

# Linearization basics

- Linearization is Taylor expanding a function about a point and ignoring the higher order terms
- We do the expansion about a fixed point
  - question - can we expand about any point and still linearize a system?

## Taylor expansion

Taylor expand  $\dot{x} = f(x, u)$  to approximate  $f(x + \Delta x, u + \Delta u)$

with  $\Delta x = x - x_0, \Delta u = u - u_0$

$$\dot{x} = \frac{d}{dt}(x_0 + \Delta x) = \frac{d}{dt}\Delta x, \text{ OR } f(x, u) = \frac{d}{dt}\Delta x$$

$$\begin{aligned} f(x, u) \Big|_{x_0, u_0} &= f(x_0, u_0) + \frac{\partial f}{\partial x} \Big|_{x_0, u_0} \cdot (x - x_0) + \frac{\partial f}{\partial u} \Big|_{x_0, u_0} \cdot (u - u_0) + \\ &\quad \frac{1}{2!} \frac{\partial^2 f}{\partial x^2} \Big|_{x_0, u_0} \cdot (x - x_0)^2 + \frac{1}{2!} \frac{\partial^2 f}{\partial u^2} \Big|_{x_0, u_0} \cdot (u - u_0)^2 + \dots \end{aligned}$$

Since linearization is to be valid in a small zone around the *fixed points*

$$(x - x_0)^n \approx (u - u_0)^n \approx 0 \quad \forall n \geq 2$$

$$f(x_0, u_0) = 0$$

Hence we can say

$$\dot{\Delta x} \Big|_{x_0, u_0} = \frac{\partial f}{\partial x} \Big|_{x_0, u_0} \cdot (x - x_0) + \frac{\partial f}{\partial u} \Big|_{x_0, u_0} \cdot (u - u_0)$$

$$\dot{\Delta x} = \frac{Df}{Dx} \Delta x + \frac{Df}{Du} \Delta u$$

$$\dot{\Delta x} = A \Delta x + B \Delta u$$

It is customary (abuse of notation) to drop the  $\Delta$ . Hence,  $\dot{x} = Ax + Bu$

$D$  is the Jacobian matrix if  $x$  and  $u$  are vectors (system of equations)

$$\frac{Df}{Dx} = \begin{bmatrix} \frac{f_1}{x_1}, \frac{f_1}{x_2} \\ \frac{f_2}{x_1}, \frac{f_2}{x_2} \end{bmatrix}, \quad \frac{Df}{Du} = \begin{bmatrix} \frac{f_1}{u_1}, \frac{f_1}{u_2} \\ \frac{f_2}{u_1}, \frac{f_2}{u_2} \end{bmatrix}$$

## Example

## propeller mechanical model

$$J\dot{\omega} + b(\omega)\omega^2 = T_{aero}$$

$$\dot{\omega} = -\frac{b(\omega)\omega^2}{J} + \frac{T_{aero}}{J}$$

$J$  = properller inertia ( $\text{kg} \cdot \text{m}^2$ )

$b(w)$  = aero damping coefficient ( $\text{Nm} \cdot \text{s}^2$ )

$\omega$  = speed of the propeller( $\text{rad/s}$ )  $\leftarrow x$  (state)

$T_{aero}$  = aero resistive torque ( $\text{Nm}$ )  $\leftarrow u$  (input)