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```
function [a,inc,W,w,e,th] = OrbitalElements( el )

% Extract the individual elements from a vector of orbital elements.
%
% Inputs:
%   el          Orbital elements vector [a,i,W,w,e,th]
%
% Outputs:
%   a           Semi major axis          [km]
%   inc         Inclination              [rad]
%   W           Right ascension          [rad]
%   w           Argument of perigee      [rad]
%   e           Eccentricity
%   th          True anomaly             [rad]
%

a = el(1);
inc = el(2);
W = el(3);
w = el(4);
e = el(5);
th = el(6);
```

*Published with MATLAB® R2019b*