

---

```
function th = TrueAnomFromEccAnom( E, e )
%
% Compute the true anomaly from the eccentric anomaly, for eccentric
% orbits
%
% Inputs:
%   E      Eccentric anomaly (rad)
%   e      Eccentricity (0 <= e < 1)
%
% Outputs:
%   th     True anomaly (rad)
%
th = 2*atan( sqrt((1+e)./(1-e)) .* tan(E/2) );
```

*Published with MATLAB® R2019b*