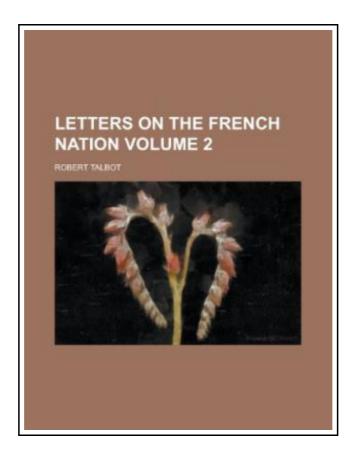
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RareBooksClub. Paperback. Book Condition: New. This item is printed on demand. Paperback. 36 pages. Original publisher: Hampton, VA: National Aeronautics and Space Administration, Langley Research Center; St. Louis, MO: US Army Aviation Systems Command, 1992 OCLC Number: (OCoLC)60315501 Excerpt: . . . 3. 0 Computational Results Aerodynamic sensitivity derivatives computed using the incremental formulation, Eqs. (27) and (28), are presented for two laminar example problems, and are compared with the same results reported in Ref. 23 for the identical example problems. In Ref. 23, these same sensitivity derivatives were computed using direct solver based methods applied to the standard formulation of Eq. (13). It is significant to note that at the outset of this study all attempts to solve these sensitivity equations in standard form using a conventional line Gauss-Seidel iteration method (Refs. 29, 30) for these two example problems diverged, despite efforts to force convergence through the use of successive line under-relaxation. This failure is attributed to the ill-conditioning of the coefficient matrix. 3. 1 Internal Flow-Double-Throat Nozzle Problem The first example problem is that of an internal flow through a double-throat nozzle, where the flow is accelerated from a Mach number on the inflow boundary of about 0. 10, to a Mach number which exceeds 2. 80 at some places on the outflow boundary. The Reynolds number, REL, is 100, based on a reference length, L, of one-half the height of the nozzle at the smaller of the two throats. Figure 1 illustrates the geometry and computational grid which is used, and Fig. 2 depicts the Mach contours of the steady-state solution; both of these figures are taken from Ref. 23, where more complete information is given. Other studies have been conducted involving the numerical computation of...



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