

Airfoil Tools

Search 1638 airfoils



You have 0 airfoils loaded.
Your Reynold number range is 50,000 to 1,000,000. ([set](#))

Search

NACA 2412 (naca2412-il)

NACA 2412 - NACA 2412 airfoil



Details
(naca2412-il) NACA 2412
NACA 2412 airfoil
Max thickness 12% at 30% chord.
Max camber 2% at 40% chord
Source [UIUC Airfoil Coordinates Database](#)
[Source dat file](#)
The dat file is in Selig format

Dat file
NACA 2412
1.0000 0.0013
0.9500 0.0114
0.9000 0.0208
0.8000 0.0375
0.7000 0.0518
0.6000 0.0636
0.5000 0.0774

Parser
No parser warnings

[Send to airfoil plotter](#)
[Add to comparison](#)
[Lednicer format dat file](#)
[Selig format dat file](#)

Similar airfoils

E207 (12.04%)	Preview	Details
E220 (11.48%)	Preview	Details
S8055 (12%)	Preview	Details
NACA CYH	Preview	Details
RAF 38 AIRFOIL	Preview	Details
LDS-2 AIRFOIL	Preview	Details
MH 120 11.57%	Preview	Details
GOE 704 AIRFOIL	Preview	Details
ONERA OA212 AIRFOIL	Preview	Details
NACA 1412	Preview	Details

Polars for NACA 2412 (naca2412-il)

Plot	Airfoil	Reynolds #	Ncrit	Max Cl/Cd	Description	Source
<input checked="" type="checkbox"/>	naca2412-il	50,000	9	32.5 at $\alpha=7.25^\circ$	Mach=0 Ncrit=9	Xfoil prediction Details
<input type="checkbox"/>	naca2412-il	50,000	5	34.6 at $\alpha=6.5^\circ$	Mach=0 Ncrit=5	Xfoil prediction Details
<input checked="" type="checkbox"/>	naca2412-il	100,000	9	50 at $\alpha=6.75^\circ$	Mach=0 Ncrit=9	Xfoil prediction Details
<input type="checkbox"/>	naca2412-il	100,000	5	49.4 at $\alpha=6^\circ$	Mach=0 Ncrit=5	Xfoil prediction Details
<input checked="" type="checkbox"/>	naca2412-il	200,000	9	66.6 at $\alpha=6^\circ$	Mach=0 Ncrit=9	Xfoil prediction Details
<input type="checkbox"/>	naca2412-il	200,000	5	62.6 at $\alpha=5.25^\circ$	Mach=0 Ncrit=5	Xfoil prediction Details
<input checked="" type="checkbox"/>	naca2412-il	500,000	9	87.3 at $\alpha=5^\circ$	Mach=0 Ncrit=9	Xfoil prediction Details
<input type="checkbox"/>	naca2412-il	500,000	5	78.3 at $\alpha=4^\circ$	Mach=0 Ncrit=5	Xfoil prediction Details
<input checked="" type="checkbox"/>	naca2412-il	1,000,000	9	101.4 at $\alpha=4.5^\circ$	Mach=0 Ncrit=9	Xfoil prediction Details
<input type="checkbox"/>	naca2412-il	1,000,000	5	87 at $\alpha=4.5^\circ$	Mach=0 Ncrit=5	Xfoil prediction Details

Update plots

[Reynolds number calculator](#)

Set Reynolds number and Ncrit range

Update Range

Reynolds Number

Low

50,000

High

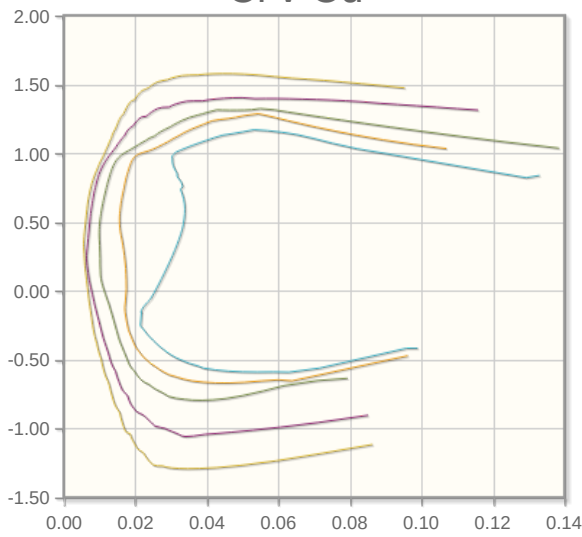
1,000,000

NCrit

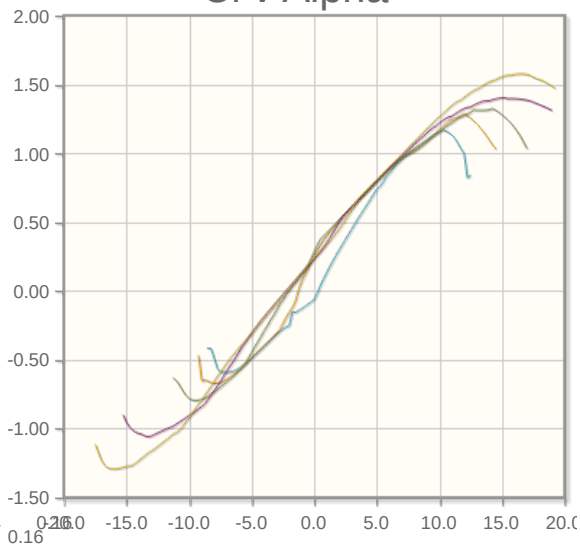
7

9

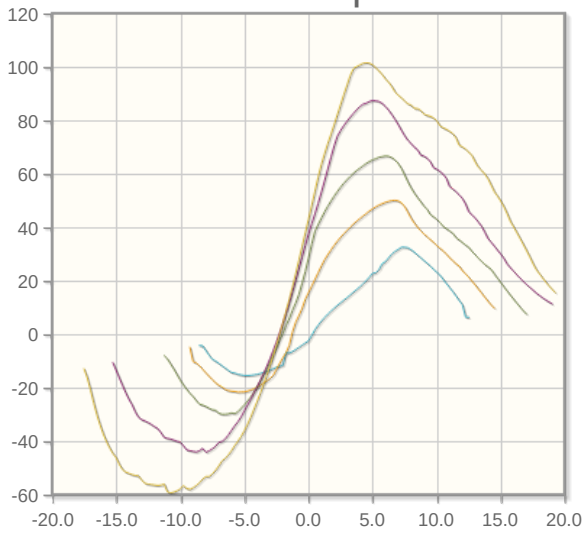
Cl v Cd



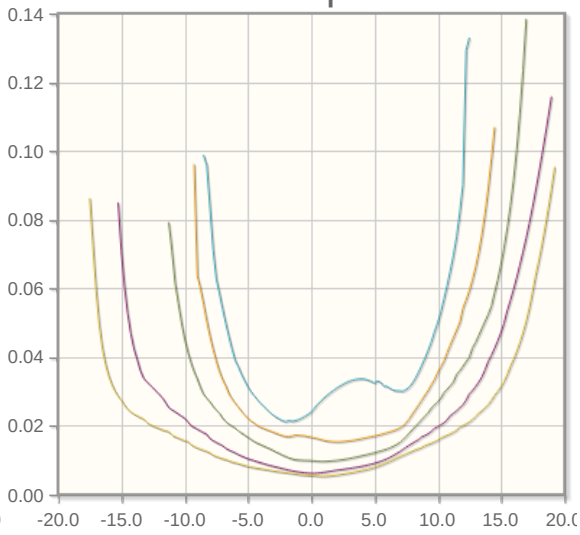
Cl v Alpha



Cl/Cd v Alpha



Cd v Alpha



Cm v Alpha

