End-to-end ground calibration and in-flight performance of the FIREBall-2 instrument

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Abstract. The payload of the Faint Intergalactic Redshifted Emission Balloon (FIREBall-2), the second generation of the FIREBall instrument (PI: C. Martin, Caltech), has been calibrated and launched from the NASA Columbia Scientific Balloon Facility in Fort Sumner, New Mexico. FIREBall-2 was launched for the first time on the September 22, 2018, and the payload performed the very first multi-object acquisition from space using a multi-object spectrograph. Our performance-oriented paper presents the calibration and last ground adjustments of FIREBall-2, the in-flight performance assessed based on the flight data, and the predicted instrument's ultimate sensitivity. This analysis predicts that future flights of FIREBall-2 should be able to detect the HI Lyα resonance line in galaxies at $z \sim 0.67$, but will find it challenging to spatially resolve the circumgalactic medium. © *2020 Society of Photo-Optical Instrumentation Engineers (SPIE)* [DOI: 10.1117/1.JATIS.6.4.044004]

1 Introduction

The circumgalactic medium (CGM) plays a critical role in the evolution of galaxy discs, as it hosts important mechanisms regulating their replenishment through inflows and outflows.[1] Directly mapping the HI Lyman alpha (Lyα) emission from circumgalactic gas at high redshifts ($z > 2$) is an insightful alternative to absorption spectroscopy, providing a new perspective with a complete 2-D or 3-D mapping of the gas distribution.[2–7] Its importance notwithstanding, emission data are very scarce in the 10-billion-year span from $z \sim 2$ to the present because of the difficulties inherent to vacuum UV observations.

The Faint Intergalactic Redshifted Emission Balloon (FIREBall-2) instrument, jointly funded by CNES and NASA, has been developed to help fill this gap and pave the way for future orbital projects. It has been optimized to provide a bi-dimensional (x, λ) map of the extremely faint diffuse Lyα HI emission in the CGM at $z \sim 0.7$ and to observe around 200 targets in a single night's flight.

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FIREBall-2 is designed to have a wide field of view (FOV) of 20.5×37 arcmin^2, a sharp angular resolution of ~6 arc sec FWHM, and a resolving power of ~1500 for a diffuse object. Given these instrument design choices, FIREBall-2 has the capability to make direct measurements and set statistical constraints on the mass, distribution, and velocity of multi-phase gas in the CGM. FIREBall-2 targets multi-object high-resolution spectra in order to simultaneously address the following science goals: measuring the total energy budget of the CGM, studying the gas distribution in galaxy halos as a function of galaxy type, constraining inflow and outflow mechanisms, deriving budgets for the exchange of mass, metals, and energy between galaxies and their surroundings, as well as many other applications.

The instrument is a balloon-borne 1-m telescope coupled to a UV multi-object spectrograph (MOS) designed to image the CGM in emission. Its goal is to target the Lyα line redshifted in the stratospheric balloon flight's spectral window near 200 nm for the universe at $z \sim 0.7$.

The overall science requirements set ambitious technical goals within a budget of a stratospheric program in terms of detector, optics, and guidance. The instrument design choices along with their technical rationale are summarized in Table 1.

In this paper, we describe the end-to-end ground calibration and in-flight performance of the FIREBall-2 instrument. In Sec. 2, we describe the instrument design and summarize the last ground adjustments of FIREBall-2. The reduction pipeline of the flight data is described in Sec. 3 and the in-flight performance assessed based on the available flight data is described in Sec. 4. As FIREBall-2 has been funded for two other flights, we will explore in Sec. 5 the extrapolated instrument's optimal sensitivity based on flight and post-flight data.

Table 1 FIREBall's design summary.

Parameter	Value	Rationale
Telescope design	1.25-m moving flat siderostat. Fixed $f/2.5$ paraboloid (1.0 m)	Siderostat-based pointing system reduces moving parts inertia and total cost, $f/2.5$ maximizes the FOV and diffuse sensitivity
Focal corrector	$f/2.5$ two mirror system	Provides a corrected 0.6-deg FOV
Spectrograph	Fast UV multi-object spectrograph with an aspherical grating	More efficient than integral field spectrograph (IFU) for the low density bright enough targets
Detector	Delta-doped EMCCD in photon-counting mode	Read noise ~0 and increased QE (~50%) for photon-starved astronomy
Guiding system	33 Hz – 1-m pupil in visible	Guide with $\epsilon_{\text{Tracking}} < 1''$ rms/axis on stars <12th mag
Wavelength range	199 to 213 nm	Stratospheric UV window
Cooling system	Ice fusion and water boiling system	Dissipate 35 MJ in the $[-60, +40]°C$ range of environment T°
Mask system	Four science masks	Adapted to observation time
	Five calibration masks	Required for calibration procedures
Spatial resolution (FWHM)	~6$''$ in both axis as the PSF is hardly affected by the slit width	Resolve CGM emission at $z \sim 0.7$ and separate it from disk emission
Resolving power	$R_{\text{diffuse}} \sim 1500\lambda/d\lambda$,	Resolve gas kinematics (>10^2 km/s)
	$R_{\text{Pointlike}} \sim 2000\lambda/d\lambda$	
Slit dimensions	slit$_{\text{dim}} \sim 6'' \times 24''$	Target CGM extension
	slit$_{\text{width}} \sim 1.5$ Å	Based on SNR optimization

2 Instrument Overview and Calibration

2.1 FIREBall-2 Design Summary

The FIREBall-2 gondola has a height of ~5 m and a weight of ~2.3 metric tons (including the 500-kg ballast).[8] This payload is designed to reach an altitude of 35 km attached to a stratospheric balloon filled with 1.1 million cubic meters of helium. Because the atmosphere absorbs most of the UV light coming from the cosmos, the FIREBall spectrograph has been optimized to operate in the narrow atmospheric transmission window around 2000 Å, where there is a dip in the atmospheric UV absorption above 35 km.[9] The nominal UV bandpass for this window (1990 to 2130 Å) transmits nearly half of the incident UV light and the residual continuum atmospheric background is comparable to or lower than the extragalactic UV background. Within this atmospheric window, there is a contaminating emission band between 2044 and 2062 Å from NO-δ, which is emitted at altitudes well above that of the balloon, but the zodiacal contribution is negligible.[10] The typical atmospheric transmission is 50% when the balloon is at float altitude and the pointing has an elevation of 50 deg. In the subsequent modeling, we have assumed a total sky background (extragalactic + atmospheric) of 500 continuum units ($n_{\mathrm{photon}} \cdot \mathrm{s}^{-1} \cdot \mathrm{cm}^{-2} \cdot \mathrm{sr}^{-1} \cdot \text{Å}^{-1}$).

The optical design of FIREBall-2 relies on a low-inertia 1.2-m sidereostat mounted on a controlled gimbal system that stabilizes the reflected beam in the gondola's frame and directs it to a fixed $f/2.5$ paraboloid that in turn focuses it at the entrance of the $f/2.5$ instrument (see Fig. 1). The object selection is achieved with a series of pre-installed precision slit mask systems that also feed the guider camera (see Fig. 2). The detector is a "Teledyne-e2v" electron multiplying CCD (EMCCD) anti-reflection-coated and delta-doped by the Jet Propulsion Laboratory (JPL).[11]
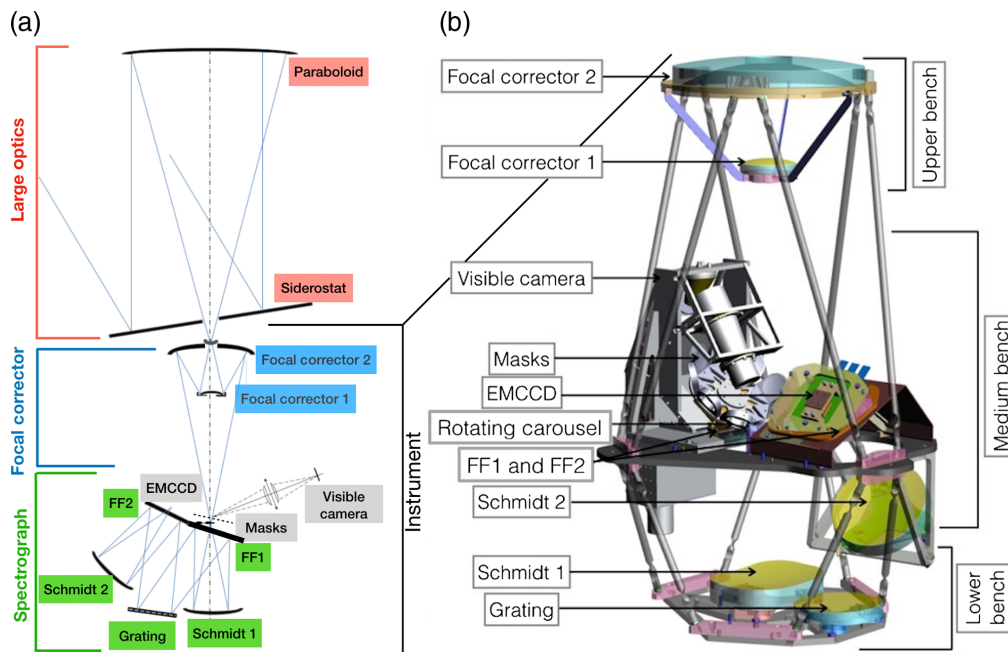

Fig. 1 (a) FIREBall-2 optical design layout. For readability, the instrument is shown in the payload layout as a flattened, plane optical layout, rather than the compact 3-D design in the real world. (b) Interior of the FIREBall-2 spectrograph. Note that it is rotated about 180 deg along the z axis compared to the left side in order to improve the visualization of the different components. During flight, this equipment is in a vacuum tank kept at ~10⁻⁶ mB by cryopumping and cryosorbtion at ~100 K, which reduces heat losses in the detector cooling chain by minimizing ice trapping, and cools the detector near 170 K. The whole vacuum tank is placed on tip–tilt actuators in order to control the focus and centering during the flight.
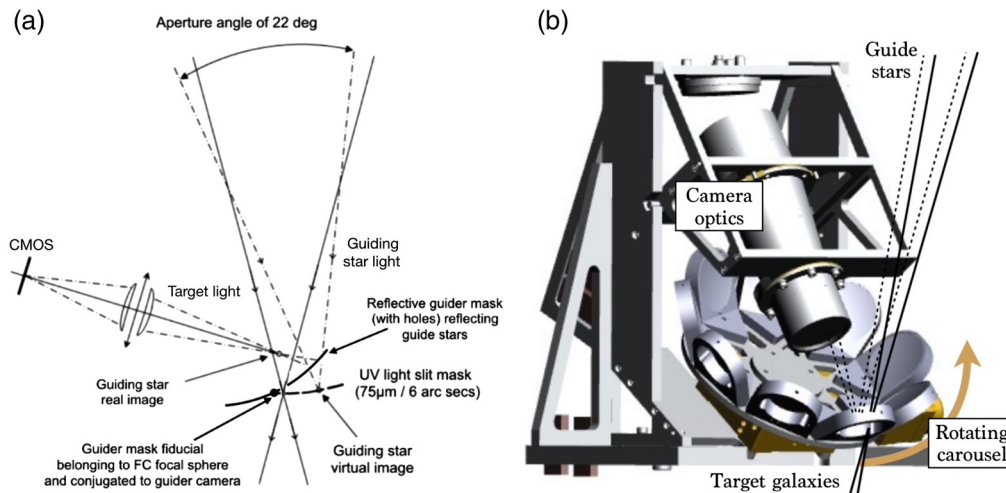
Fig. 2 (a) Mask-guider-system description scheme. The guider mask is a reflecting mirror, just above the science mask, with holes at the target locations to let the light enter the spectrograph. The guiding specification is 1 arc sec rms per axis for X and Y, but the hanging ground test on a bench simulating the complete flight chain have achieved a few tenth of arc-seconds.[15] The third guidance axis is provided by a rotating stage moving the complete payload guided to 1 arc min so that its contribution to the image is small. (b) Each mask unit, consisting of a guider mask attached to a science mask, is mounted on a rotating carousel and confocal within ± 50 μm. Dotted lines represent light from guide stars, which are reflected, the solid lines are target galaxies, which pass through the slit mask.

The 1-m telescope shows very good performance at a cost compatible with suborbital programs achieved using an innovative thin mirror/hierarchical-kinematic support and telescope pointing control.[12] The telescope is mounted on a low-mass gondola made of aluminum and carbon fiber. This gondola provides four-axis guidance (a coarse azimuth control for the gondola to face the field, fine elevation, and cross-elevation to target the exact center of the field despite gondola pendulum, and field rotation) over a range of 40 deg to 70 deg in elevation. During the 2018 flight, in degraded conditions (see Sec. 3.1), the pointing stability performance on each axis was ~0.5 arc sec.

Inside the tank, the narrow FOV of the parabola alone is extended to ~800 arc min^2 by a Schwartzchild-type two-mirror field corrector $f/2.5 - f/2.5$ (Fig. 1). The corrector is designed to introduce a field curvature that cancels the native curvature of the Schmidt-based spectrograph to produce a flat focal surface at the detector. The slit masks of the spectrograph have been fabricated at the same curvature and are mounted on a rotation stage. The spectrograph itself consists of identical $f/2.5$ Schmidt collimator and camera mirrors, the aspheric correction being provided by the 2400 g/mm reflection grating obtained by double replication of a deformable matrix using the Lemaitre method.[13] Flat folding mirrors are used to reduce the volume of the vacuum tank in which all the optics and detector are encased. The Invar spectrograph structure and the Zerodur™ low-expansion glass make the instrument robust against temperature changes. A layout of the overall payload and the instrument are, respectively, presented in Fig. 1.

In addition to the science objectives described in Sec. 1, the flight of the FIREBall-2 instrument also validated many processes and state-of-the-art hardware components (UV EMCCD,[14] CNES pointing system,[15] cooling system, and large UV aspherical grating[13]).

2.2 Main Ground Alignment

The FIREBall-2 instrument complexity and the intricacy of its sub-components require a hierarchic adjustment and calibration procedure. An especially demanding specification is the absolute guidance to 1 arc sec (~1/6 slit width) during the ~2 hours of observation per field. This goal is made still more challenging by the slight mask-to-mask shape differences (see Sec. 2.1) that require a mask-dependent calibration.

The optical fine adjustment procedure has been split into two major blocks, each containing several alignment procedures.

2.2.1 Internal adjustments

This section details the five steps required to align all subsystems inside the tank (sequentially: masks in the mask wheel → mask wheel in the guider system → guider system, spectrograph, FC, and detector on their bench) and place them at their nominal position in the medium bench mechanical reference (see Fig. 1 right).

1. Relative positioning of masks to their nominal location with metrology feedback and iterative shimming. The final measurements using a contactless STIL distance chromatic sensor show that all of the science masks are confocal within less than ± 50 μm.

2. Positioning of the mask wheel at a location that ensures the right platescale with Faro Arm feedback.

3. Spectrograph and focal corrector alignment performed with Faro Arm feedback using a mechanical reference on the medium bench. From now on the spectrograph and focal corrector assembly will be designated as "the instrument" (Fig. 1).

4. Centering and focus of the guider. The fiducial on the guider mask is conjugated with the guider camera (see Fig. 2). Illuminated with visible light, this fiducial is positioned at the central pixel of the guider camera.

5. Detector focusing with feedback from full-field analysis of the resolution of the UV spectra of a Zn emission line lamp produced by a full-pupil diffuse illumination of the science and calibration masks. To infer the position of the focal plane, a tilted calibration mask that crosses the focus in both directions has been used initially and then refined directly on each science field image. The final detector focus has been optimized to balance the image quality in the four science fields, so that the residual defocus is that generated by the scatter of the mask positioning ($\sim$25 μm rms).

2.2.2 Full instrument alignment

This four-axis conjugation (two axis-centering operations and two tip–tilts) aims at aligning the instrument (spectrograph + FC) with the well aligned (steps 2 and 3 below) paraboloid and siderostat. After the instrument is aligned and centered on the paraboloid axis, a final through focus is to be achieved in flight to compensate for thermoelastic deformations. The ground conjugation is achieved in four steps.

1. Centering the instrument (tank) on its platform at the middle of its tip–tilt actuators, floor translation ranges and nominal elevation shims in order to have margin for the final alignment. We place a mirror on the upper part of the focal corrector assembly (called "FC mirror" in the next steps).

2. Aligning the parabola (centering and alignment, respectively, at $\sim$1 mm and $\sim$0.2-deg precision) with the FC mirror by adding an orthogonal laser centered in the parabola hole. Center direct and reflected laser spots at the instrument top ensure the parabola has been roughly set with its axis close to that of the instrument.

3. Finding the values of the sidereostat tip–tilt encoder position where it is orthogonal to the parabola axis (autocollimation). A movable $f/2.5$ visible light upward source is placed near the parabola focal point and adjusted along with the sidereostat orientation to minimize the coma of the return image. A byproduct of this operation is a good indication of the focal point location.

4. Final and accurate centering of the instrument. The siderostat is set to autocollimation position (cf 3) and the mask is illuminated with visible light. The tip–tilt focus mechanism is adjusted to superimpose in the guider the direct and returned images of the illuminated mask.

2.2.3 *Guider mapping and alignment*

Such a balloon-borne multi-object spectrograph is at the limit of complexity of what can be calibrated at a launch base with extremely limited ground support equipment (GSE). It requires very demanding plate-scale verification (~0.5%) and absolute *X-Y* in-flight positioning (~1″) of the targets because their masks must be cut before the launch. Indeed, a major challenge of the FIREBall-2 project was to develop a 100% self-consistent method that does not require any GSE for all of the pre-flight adjustments and calibration of a space facility. To this end, an invaluable asset of FIREBall-2 is its siderostat, which provides a full-pupil autocollimation capability. This is made possible by a good enough atmospheric transmission at 200 nm up to distances of a few tens of meters at ground level.

The goal of the *XY* calibration is to calibrate the science mask centering relative to the guider and determine on which exact pixels in the guider the guide stars should be placed to ensure that each target UV light is well centered into the 6-arc sec slits (~80 μm). This must be achieved at arcsecond precision, which prevents the use of a collimator much smaller than the primary mirror because the residual aberrations would bias the calibration.

To this end, the full-pupil autocollimation (see Fig. 3) mode has been used extensively in this calibration in order to avoid sub-pupil effects on image quality. The siderostat encoders are first calibrated against star images acquired with the guider at ~0.2% accuracy. Note that this calibration is independent of the guider distortion. Then by putting a fiber illuminated by a Zn lamp (emission lines at $\lambda = 202.61371$, 206.26604, and 213.92635 nm under vacuum) at the focal plane of the 1-m paraboloid (facing up) and by moving the siderostat, we are able to perform any desired angular move of the observed spot anywhere in the UV detector plane (without a mask) and the guider plane (for each mask), independently of the dispersion.

This capability is used to attach the pixel position of the guider stars to the targets position in the sky. A detailed description of this method is beyond the scope of this paper.

2.2.4 *Focusing*

The autocollimation mode has also been used to achieve and assess a 5-arc sec image quality over the 0.6-deg UV FOV using the full $f/2.5$ aperture ratio to maximize precision.
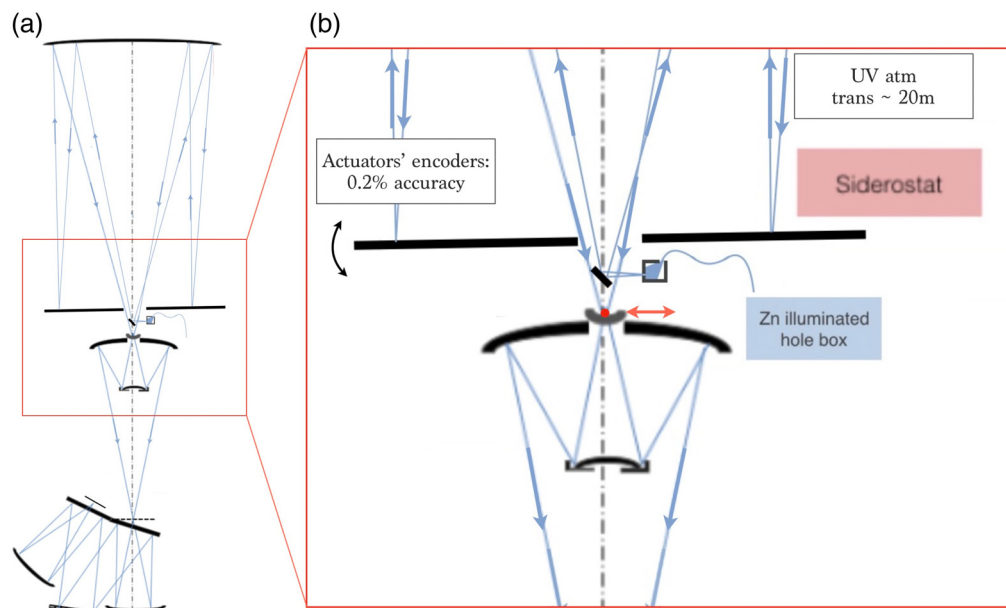

Fig. 3 (a) FIREBall-2 instrument in full-pupil autocollimation mode. (b) Zoom on the actuated part. Moving the siderostat actuators with high accuracy (black arrow) allows to move the image of the fiber (red dot on the siderostat focal plane) all over the detector plane while staying in full-pupil illumination, which is necessary to avoid sub-pupil effects that would improve artificially the PSF of the large optics.

Because only the guider images can be used to focus the instrument while in flight, the calibration procedure has provided the offset between the guide stars focus in visible light in the guider, and that of the targets in UV at the spectrograph detector. To this end, series of through-focus have been performed at different positions in the field with the same setup as in Sec. 2.2.3. In flight, the best UV focus is thus reached by setting the instrument to the calibrated defocus of the guide stars.

2.3 Pre-Flight Performance

2.3.1 Instrument resolution

Spectral resolution. The spectrograph spectral and angular point spread function (PSF) were measured before the instrument integration. This was performed using a holographic UV diffuser covering the whole UV field with an $f/2.5$ aperture and fed by a Zn emission line lamp. The diffuser was set near the focal surface at the payload entrance and was found to provide a uniform illumination to better than a factor 2. It illuminates in turn each science mask so that the image at the detector shows well-separated images of the three UV Zn lines over the whole field. The detector was then shimmed iteratively to analyze the evolution of the PSF at three wavelengths (Sec. 2.2.3) with the focus. After integration, the spectrograph resolution was re-evaluated for each science mask by applying simple de-convolution algorithms. This technique measures both the dimensions of the slits and the resolution of spectrograph. The results for each mask are shown in Fig. 4. For a point source object, the spectral resolution of the spectrograph is found to be ~0.5 Å FWHM ($R \sim 2800\lambda/d\lambda$). As the PSF at the science mask level is smaller than the slit's width, adding quadratically the angular resolution of the guider, the large optics, and the focal corrector (see below) yields to an overall spectral resolution for a point source object of $R \sim 2000\lambda/d\lambda$.

For diffuse objects, the spectral resolution is deteriorated by the spectral dimension of the slit. This leads to a $\lambda/d\lambda$ spectral resolution between 1300 and 1600 for the two sets of masks, allowing the spectral separation of components with a velocity difference larger than 200 km/s. The mean dispersion is found to be ~608 ± 2 μm/nm in accordance with the ZEMAX optical model, which predicts ~606 ± 0.5 μm/nm.[16]

Spatial resolution. The resolution of each sub-system of the instrument has been evaluated independently using full-pupil illumination and applying a deconvolution algorithm.

The contribution to the spatial resolution of the guider, the large optics, the focal corrector, and the spectrograph are, respectively, 1.8″, 1.8″, 2.4″, and 2.8″ FWHM (see Fig. 5).
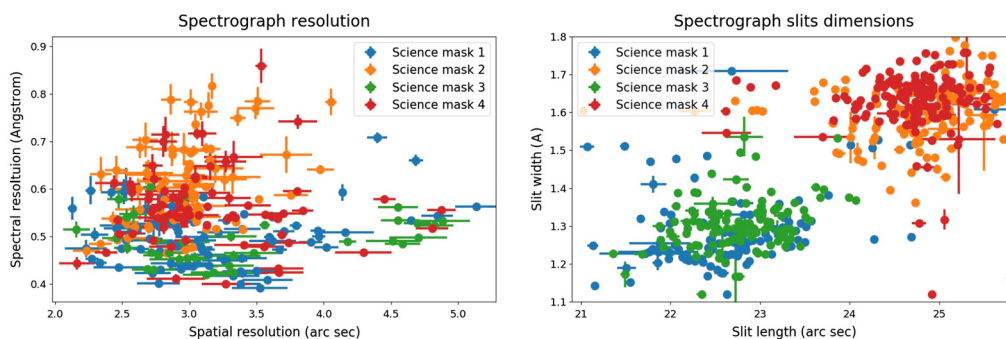

Fig. 4 (a) Spatial and spectral resolution FWHM of the spectrograph alone for a point source object. For diffuse objects, the spectral resolution must be convolved by the ~1.5-Å slit width. The resolution's scatter is mostly due to the shape of the masks. As the slits are located between the focal corrector and the spectrograph where the focal plane is spherical, the slits have been cut on a sphere. Some masks depart slightly from this nominal sphere which generates a slight defocus across the FOV. (b): Slits' spatial and spectral dimensions for the four science masks. Two sets of masks are apparent, corresponding to a manufacturing with a different laser cutting setup.
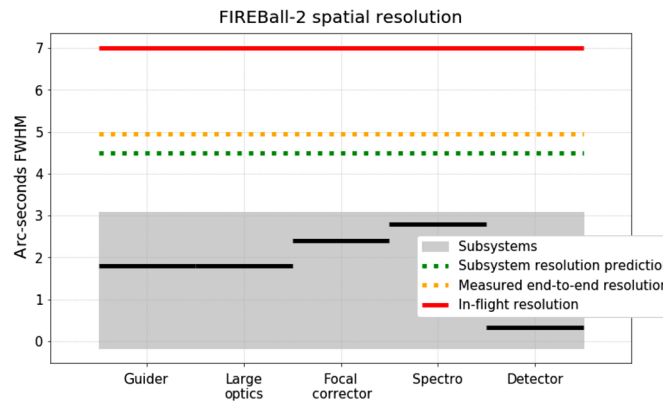
Fig. 5 Contribution of the different subsystems to the FWHM spatial resolution in arcseconds (black lines). The degradation due to the pixels' size (∼1.1″/pix) is negligible and taken into account in the resolution measurement performed on the flight-detector images. The conservative end-to-end measurement in autocollimation is plotted in red. The final prediction of the instrument in-flight resolution is plotted in orange and the flight value in red.

The convolution of the different optical sub-systems yields an FWHM of ∼4.5″. In parallel, end-to-end images have been taken with an 8-arc sec-diameter fiber at the parabola focal plane to illuminate the instrument in autocollimation (Fig. 3): the beam reflects off the large optics (twice on the paraboloid) and then goes through the total payload (focal corrector and spectrograph). After 2-D deconvolution of the fiber disk, the image at the detector level predicts an overall spatial resolution of ∼4.9″ FWHM, which is a conservative measurement as it implies a double pass on the paraboloid and because the fiber was not yet optimally at the center of the FOV. Due to mechanical deformations and in-flight defocus, we expect a small ground-to-flight image deterioration: the science masks' Z position are known with an accuracy of ∼50 μm, which generates a 1″ FWHM degradation (tested under ZEMAX).

2.3.2 Throughput and optics transmission

A key optical component of the FIREBall-2 spectrograph design is the high-throughput cost-effective, holographic 2400 l/mm, 110×130 mm^2 aspherized reflective grating used in the range 199 to 213 nm, near the 28-deg deviation angle.[17]

The throughput of the overall instrument is measured by comparing input and output fluxes with a solar blind CsTe photocathode cell from ITT, which gives a 17% efficiency for the instrument alone.

The total FIREBall-2 throughput is found to be 11% (large optics: 65%, instrument: 17%, vignetting is included, detector quantum efficiency is not.) This is a factor of 6 above what has been observed in flight for FIREBall-1.[18,19] This has been achieved despite the more complex optics required to correct the larger FOV and double the angular resolution for this new generation of the FIREBall instrument. The throughput gain results from the 55% efficiency at 204 nm of the FIREBall-2 grating (versus 32%) and from the large losses in the FIREBall-1 fibers. The additional gain of a factor 6 on the detector quantum efficiency combined with the above and the improved guidance provides an order of magnitude gain in faint source detection efficiency to FIREBall-2.

3 2018 Flight and Data Processing

3.1 Flight

FIREBAll-2 was launched on a 1.1-million-cubic-meters balloon from the Fort Sumner Columbia Scientific Balloon Facility on September 22 at 10:20 AM local MDT. The flight sequence is detailed in the work by Hamden et al.[8]

After ~3 h of ascent and ~3 h at float altitude, the balloon began to lose altitude around sunset, most likely due to a hole in the balloon. Because of this anomaly, the balloon dropped to its minimal mission requirement altitude after 45 min of science acquisition instead of the >6 h required. Balloon altitude is important for science as between 30 and 40 km altitude, each km of altitude loss represents a reduction of transmission of ~5%.

In addition, weather constraints resulted in a launch the day before a full moon (moon 94% full during the flight). The deshape of the balloon due to the balloon's deflation also impacted science data by concentrating a specular reflection of the moonlight over the balloon directly at the entrance of the payload. This boosted our background levels by >1000 times what was expected. The resulting scattered light produced ~1 event/pix/50 s frame background, nearly an order of magnitude too high to efficiently perform photon counting.[20]

Aside from these considerations, almost all subsystems performed nominally. Some flight anomalies are detailed in the work by Hamden et al.[8] as well as the four fields observed. In order to focus on the instrument performance, we will only detail the reduction and analysis of the first field (DEEP 2: RA = 253.06, DEC = 34.97, $\theta = 20$ deg) when the instrument was at the nominal altitude.

3.2 DS9 Quick-Look and Reduction Pipeline

In order to analyze flight data in real time, we built a python FIREBall dedicated DS9 extension similar to the DS9[21] Quick Look plugin.[22] Indeed, DS9 provides easy communication with external analysis tasks and is extensible via XPA. This extension was designed for real-time processing in order to support in-flight decisions as well as assembly, integration, and tests (AIT). During the flight, it was mostly used for cosmic ray (CR) removal, background subtraction, help with object recognition, and stacking.

This extension proved its worth during the AIT and the flight, so the same DS9-oriented method was used to design the entire reduction pipeline to process data after the flight. This extension is divided into three main parts, each of which fills a specific need:

AIT: Supports assembly, integration, and test. This includes through-focus and through-slit visualization and analysis, *XY* calibration analysis, radial and encircled energy profiles, etc.

Flight reduction pipeline: Contains every part of the flight data processing: CR removal, over-scan correction, background subtraction, photon-counting processes, and image/object stacking.

Detector: Supports EMCCD characterization prior to and after the flight: gain computation from fluctuation analysis or histogram fitting, computation of the different noise sources [clock-induced charges (CIC), read-out noise, etc.] and other tests: auto-correlation, column-line correlation, smearing analysis, etc.

3.3 Data Processing

3.3.1 Cosmic ray removal

CRs and other high-energy particles from outer space produce spurious features when hitting the detector. These random and unavoidable events generate spikes that are amplified in EMCCDs such as any other photoelectron. Each impact creates a "tail" in the direction of charge transfer due to deferred and overspill charge in the serial multiplication register. The tail of the impact is highly dependent on the particle's energy. We note that each event profile on the detector has unique properties that might also depend on other characteristics, e.g., the integration time, the electron multiplication (EM) gain (and thus the degree of saturation in the high-gain register), the angle of the impact, etc.

In order to identify and remove CRs' peak and tail in the EMCCD images, we implemented in the DS9 extension a detection, identification, and removal algorithm. Detection is quite simple as any impact saturates at least one pixel on the device. The algorithm is based on a marching

square code that creates isolevels isolating CR impacts. It incorporates a basic but efficient classification code in order to apply a more or less important mask depending on the particle energy.

As we can detect up to 1000 CRs in a 100-s exposure, we had to optimize the speed performance of the detection/masking algorithm. After optimization, the process takes approximately 20 ms/impact on a 2.6-GHz Intel Core i7 processor.

We calculate a rate of $\sim$2.4 particles $\cdot$ cm^{-2} $\cdot$ s^{-1} in flight at 30-km altitude compared to $\sim$0.05 particles $\cdot$ cm^{-2} $\cdot$ s^{-1} on ground for the same CCD201 (13-μm pixel pitch) setup. Between 26 and 33 km above the ground, we find a small increase of the impact rate with altitude of about $+1\%$ per kilometer.

These CRs generate important signal-to-noise loss that affects about 0.05% of the image per impact. On the 50-s exposure science frames, this represents 25% of data that must be masked in order to be fully conservative. This is equivalent to reducing the number of effective images by 25%, which prevents FIREBall-2 from capturing images with too long exposure time. In this respect, this constraint will be used in order to estimate optimum single-frame integration time, currently estimated to be $\sim$30 to 80 s in order to maximize the signal-to-noise ratio (SNR) (see Sec. 5.2).

3.3.2 Systematics contribution removal

Before further processing, we need to remove systematic contributions in the EMCCD, especially account for the non-uniformity of the bias, and line-to-line/column-to-column variations.

The line-to-line variations seem to come from remaining charges in the serial amplifier or poor amplification register clearance that do not always empty it to the same level. Because of a higher charge transfer inefficiency for these highly saturated events, some CR tails can continue on the left over-scan region. For extremely energetic events ($\sim$0.1%), it also appears that the charges of the CRs are not cleared and then continue over several lines from the left over-scan to the right. Because of this, to estimate the bias level on each line, we take the median value of the left part of the right over-scan region (between column 2200 and 2400), just after charges have been cleared out from the serial register. This prevents including in the bias any background contribution in the spectral direction.

We then correct the column-to-column variation by subtracting for each column its median value. Then the corrected value of the pixel in row i and column j is

$$p_{i,j}^{corr} = p_{i,j} - p_{2200\tilde{<}\cdot<2400,j} - p_{i,0\tilde{<}\cdot<2069}$$

The dot in the index is a placeholder to indicate whether the mean was taken over rows or columns. We then use this notation $p_{n_1\tilde{<}\cdot<n_2,j}$ for the median value of the j'th line between column n_1 and column n_2.

Note that the subtraction of the column medians in flight images with background does not remove the potential detections, as they overlap only a few lines and are completely washed out in the subtracted medians. In addition, the signal coming from the sky and reflections in the instrument, and even more its series of vertical medians, has only low spatial frequency components, which do not affect our potential detections expected over only a few pixels.

3.3.3 Background subtraction

The moon's light reflected on the balloon (see Sec. 3.1) creates high sky background count rates of 2×10^6 line units (LU) ($n_{photon} \cdot$ s^{-1} $\cdot$ cm^{-2} $\cdot$ sr^{-1}) for diffuse reflection, which is more than 2 orders of magnitude greater than what was expected.

In addition, at very specific azimuth angle differences between the instrument, the moon, and the balloon, reflective parts in the instrument are illuminated, increasing the parasitic light 10 times higher than the diffuse reflection.

Since the detector is not solar blind and we do not see any correlation between the background intensity and the altitude over a factor 5 in atmospheric transmission, we infer that the

background comes mostly from visible light leaks. Post-flight accurate scattering analysis shows that the incoming light from the balloon was not following the nominal optical path but was reflecting from different mechanical components inside the tank.

As the background changes quickly in shape and intensity between exposures, and since the CR masking modifies significantly which exposures are retained in the summation for each pixel, the mean background value varies from pixel to pixel. Therefore, the background must be estimated and removed individually on each CR-free image.

In addition to this, there is a curved and net boundary in the image background (see Fig. 6). This break, most probably due to a component shading the bottom part of the detector, makes automatic background subtraction difficult. In order to handle it, we divide the image in two curved sub-images and perform background subtraction on each independently.

The average background level is computed iteratively within 40-pixel image bins by clipping out pixels at 3σ from the average (maximum 10 iterations) and taking the median value.

The low-resolution background image is then convolved by a 3×3 2D median filter and is then re-sized to the original data size using spline interpolation.

Because it is negligible compared to the high spurious flight background, we do not need to correct for dark current ($0.08\ e^-$/h/pix on ground).

3.3.4 *Stacking and hot pixel removal*

The images are stacked by taking the mean value at each pixel position. Indeed, as the nominal flux levels we want to detect are lower than $0.5\ e^-$/pixel, taking the median would throw away important detections.

However, some hot pixels show up after stacking the images. Because they have been identified on ground dark images, and they only spread onto 3 to 4 pixels in the charge transfer direction, we masked every 4 pixels following a hot pixel identified on ground and interpolate the values of the pixels inside the mask by applying a 2D $\sigma = 1$ pixel Gaussian filter.
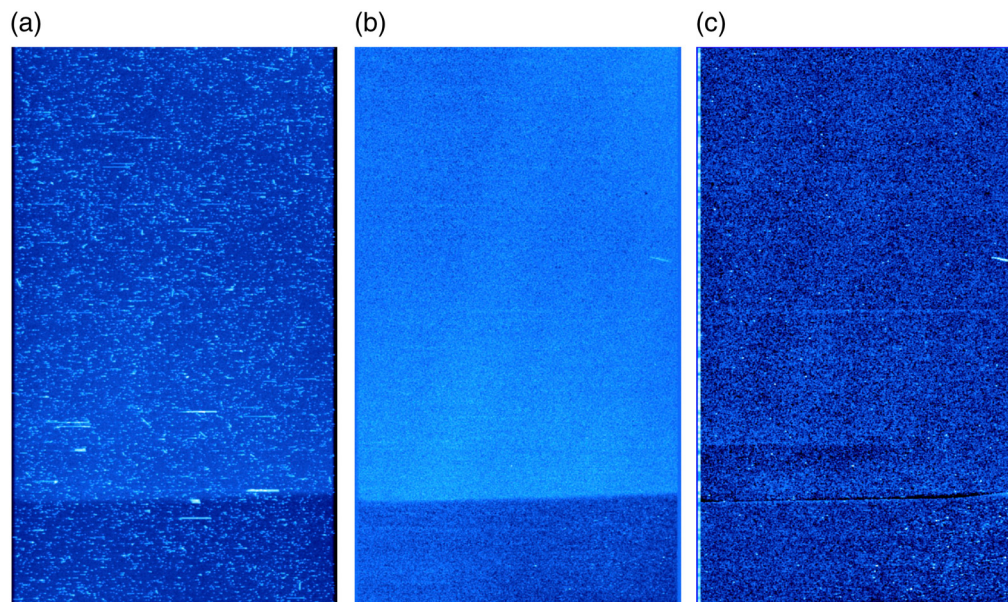
(a) (b) (c)

Fig. 6 Field 2 summed images. (a) The summed image before CR removal. (b) The summed image after CR removal. (c) The summed image after CR removal and subtracting the out-of-band background scattered light. The sharp drop-off in scattered light on the bottom fifth of the image makes the subtraction difficult, yielding a residual along the edge of the scattered region. All images have been smoothed with a Gaussian kernel with sigma of 1.5 and a radius of 3 pixels. The emission from the brightest star, Bright Star 1, is visible in the last two images as a horizontal line roughly in the middle.[8]

3.3.5 *Flux map generation*

Assessment of the amplification gain.   The gain is needed to convert the measured analog-to-digital unit (ADU) levels to photoelectrons, but careful calibration is required because of its dependency on detector temperature ($\sim$10% per degree at $T \sim -110°C$ and $G_{\rm amp} \sim 1000\ e^-/e^-$). Three methods are usually used to measure the amplification gain of the detector:

- Unique electron spectrum from a flat-field image having a low illumination level of <1 e^-/pix. Looking at the histogram of ADU values with logarithmic counts, the bias corresponds to the mode (most frequent ADU value), the readout noise to the width of the parabola around the bias, obtained through a quadratic fit, and the gain to the slope in the amplified part (highest values)[20] (see Fig. 7).
- Fluctuation analysis derived from the slope of the variance/mean ADU diagram without any correction. This gain includes the amplification process and the conversion gain.[23]
- Direct measurement by dividing two images with and without gain. The input flux must be as stable as possible and the exposure times must be different so that we stay far from the dark level for the non-amplified image and far from the saturation level for the amplified image.

Because the two first measurements can be significantly impacted if smearing appears in the CCD or in the amplification registers, the gain in flight has been post-calibrated on ground using the third method at the flight temperature. Indeed, during the flight, the smearing exponential length of $\sim$1 pixel drastically affects the amplification gain derivation from two first methods ($\sim$1400 $e^-/e^- \to \sim$650 e^-/e^-). Using method 3, we retrieve a flight amplification gain of 1370. This gain was corroborated by a complete electron spectrum model, including a toy linear model of the smearing effect, that was tested at several illumination levels.

Fluxes outside atmosphere.   We can then compute the flux in the co-add image:

$$F_{e^-/\rm pix} = \frac{\rm pix_{ADU} \times G_{conv}}{G_{amp} \times t_{exp}}$$

with $\rm pix_{ADU}$ being each pixel value in ADU, $G_{\rm conv}$ the conversion gain (1.8 e^-/pix), $G_{\rm amp}$ the amplification gain ($\sim$1370 e^-/e^-), and $t_{\rm exp}$ (50 s) the exposure time.
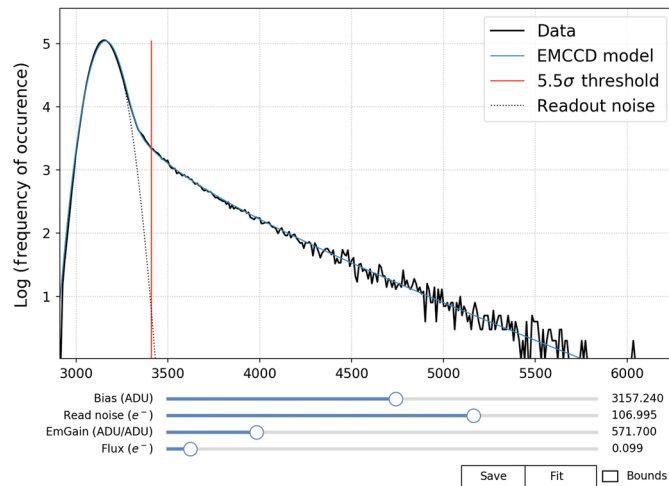

Fig. 7 Unique electron spectrum. This fitting algorithm from the FIREBall DS9 extension allows fitting and recovering EMCCD parameters in the absence of smearing. The red line represents the photo-counting threshold used to mitigate the excess noise factor due to amplification. The dotted black line represents the readout noise. In photon-counting mode, this noise is lowered to the part above the 5.5σ threshold. As explained in Sec. 3.3.5, the histogram shape is affected by the smearing that acts exactly as if the amplification gain was lower (1400 $e^-/e^- \to$ 600 e^-/e^-).

In order to retrieve the flux in photons per Angstrom at the detector level:

$$F^{\text{Det}}_{\gamma^-/\text{Å}} = \frac{F_{e^-/\text{pix}} \times \text{dispersion}}{\text{QE}}$$

where $\text{QE} \sim 50\%$ is the quantum efficiency of the detector. The dispersion of the instrument is ~ 4.66 pix/Å.

Then, above the atmosphere, the flux is

$$F_{\gamma/\text{Å}/\text{cm}^2} = \frac{F^{\text{Det}}_{\gamma^-/\text{Å}}}{A_{\text{telescope}} \times t_{\text{atm}} \times t_{\text{FB}}},$$

where $A_{\text{telescope}} = 7854$ cm^2 is the collecting area of the telescope, $t_{\text{atm}} \sim 33\%$ is the transmission of the atmosphere at the telescope average altitude, and t_{FB} ($\sim 13\%$ including large optics and vignetting) is the throughput of the FIREBall instrument.

4 In-Flight Performance

4.1 Detection and Proof of Performance

The ground-based calibration for absolute pointing and *XY* positioning described in Sec. 3 assumes a stability better than 1 arc sec of the guider-to-mask relative positioning until the end of the flight, in an environment 40°C colder than during the calibration.

The continua of the three UV-bright "GALEX" stars in the science field were detected, with an SNR ~ 18 for the brightest one ($M_{\text{FUV}} = 17.8$). This proves that the targets were positioned close to their nominal position, one of the toughest challenges of FIREBall-2. Flight data do not allow us to access the residual decentering between the source and the exact slit center. Photometric measurements are performed in Sec. 4.5.

4.2 Guidance Performance

In addition to its impact on the throughput and flight duration shown in the previous section, the loss of altitude also impacted the balloon's (and consequentially the gondola's) motion because of the varying wind direction with respect to altitude. This intensified the different gondola
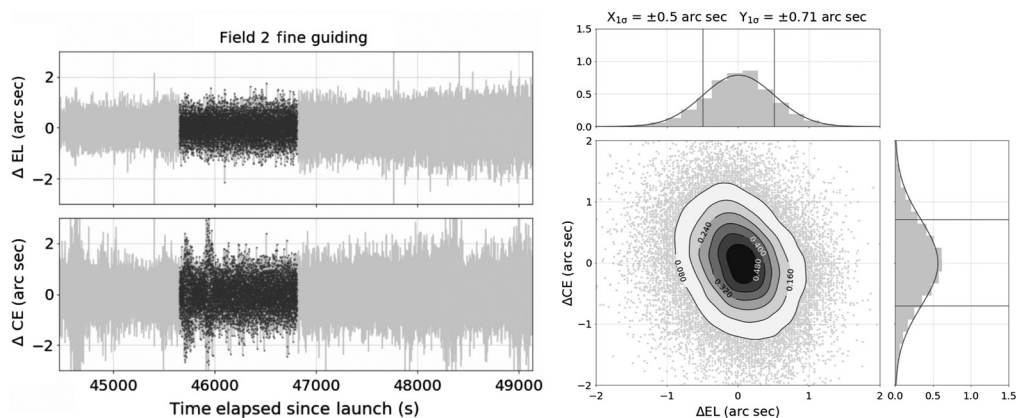

Fig. 8 (a) Evolution of in-flight fine guiding performance. (b) Summary of the two-axis (elevation and cross-elevation) fine guidance errors in closed loop on Field 2 during the ~1-h observation. Data corresponding to ballast drops have been excluded. The 1D probability density histograms are fit with a 1D Gaussian, which we use to measure the 1σ errors in the 1D distribution (vertical lines). The 2D distribution in the center is interpolated with a Gaussian kernel density estimation. The 1σ contour here corresponds to 0.39 arcsec^{-2}. FIREBall-2 achieves a (1σ) pointing of ±0.5 arc sec in elevation and ±0.71 arc sec in cross-elevation (Melso et al., in prep.).

excitation modes, requiring better damping and causing a decrease in the pointing performance. Despite that, the pointing performance stayed well within the technical specification.[8]

The in-flight raw guider error signal is found to be 0.7″ RMS per axis (Fig. 8), an upper limit to the guidance error, as it includes the guider signal noise and the field rotation guiding error of ∼0.3″ RMS. The latter number derives from a 1.5-arc min field rotation 1D guidance error, and a typical distance-guiding stars-target of 12 arc min.[15]

4.3 Image Quality

An in-flight spatial resolution of 7″ has been derived from a Gaussian fit along the spatial direction of the GALEX brightest star continuum in the first field. We are currently working at LAM on improving the focus of the spectrograph in order to obtain an in-flight 5″ FWHM angular resolution all over the field for the upcoming flight in 2021. This is of course a very important gain in resolution, and it would also decrease all the sources of noise in counts/PSF/hour as the PSF gets sharper (this is taken into account in Table 3 and detailed in Sec. 5.2).

4.4 Detector Performance

FIREBall-2 incorporates a state-of-the-art 2 K × 1 K, 13-μm-pixels electron-multiplying CCD 201 from Teledyne-e2v processed at JPL to increase its UV sensitivity with a delta-doping procedure, which yields a 100% internal quantum efficiency device limited only by the reflection losses, and a multi-layer antireflection coating to further enhance external QE.[11] The stochastic nature of the amplification process in EMCCDs generates an excess noise factor (ENF = $\sqrt{2}$ at high gain) that can be removed under certain conditions (Flux $\ll 1 \ e^-/\text{pix}/\text{exp}$ and $G_{\text{amp}} > 10 \times \text{RN}$) by applying a thresholding process known as photon counting.[20] The different characteristics of the detector are summarized in Table 2. These state-of-the-art devices require special parameter tuning in order to take advantage of their performance, as well as specific data reduction such as CR removal or photon-counting processing.

The present model of FIREBall-2 detector has been calibrated on the ground and shows a 55% peak efficiency in the nominal UV instrument bandpass and >50% in the entire bandpass. This value is compared to flight data in Sec. 4.5. The detector performance is detailed further in the work by Kyne et al.[14]

Relevant to the present analysis, the detector shows sub-optimal charge transfer efficiency (CTE) due to the unusually low temperature selected to minimize the dark current. We think this can be improved by further optimization of the controller/parameter trade-off for the next flight.

Table 2 Flight UV EMCCD characteristics.

Parameter	Unit	Value
Device's temperature	T (°C)	−110
Number of amplification pixels		604
Amplification	ADU/e^-	1400 ± 30
Bias	ADU	1.4×10^3
Read noise (no photon counting)	e^-	107 ± 1
CIC[a]	e^-/pix/frame	0.005 ± 0.002
Dark current[a]	e^-/pix/h	0.08 ± 0.03
Smearing exponential length	pix	0.7 ± 0.1
Exposure time	seconds	50

[a]Parameters measured on ground as they require extremely low illumination levels not obtained during the flight. The CIC we give here is a lower limit as we do not take into account horizontal CIC.

At flight temperature, this average CTE induces a charge smearing that yields a performance similar to an electron multiplying (EM) gain three times lower. This increases the noise contribution and causes event loss when attempting to perform photon-counting thresholding. This effect combined with the high background prevented us from using photon counting. Smearing inversion algorithms can be applied, but there is an intrinsic limitation as these inversions always generate noise amplification.[14]

4.5 *2018 Flight Sensitivity*

The total efficiency was computed based on the flux measured on Bright Star 1 using a fitting-based photometric algorithm. The measured continuum was found to be 30% lower than expected from the ground-based calibration and atmospheric model, which is well within statistical fluctuations derived from applying the same photometric algorithm at random locations. The main contributors to this uncertainty are: target centering, photometric measurement, atmospheric transmission model, focus, instrument throughput, and detector calibration. In addition, the photometric measurement on BS1 shows a decreasing flux during the observation, which result probably from defocus or target decentering rather than atmospheric model bias.

No emission line has been detected at a statistically significant level, neither by searching at the predicted individual emission line locations nor by stacking the target galaxies or a subset of the brightest ones. In order to measure the sensitivity limit for the total Lyα emission line of a target, we have performed profile fitting of a 2D Gaussian matched to the PSF at random locations in the image. We avoided areas where target signal can be expected. The derived 5σ sensitivity limit for an unresolved $z \sim 0.7$ source for the available 30-min observation is $\sim 1.5 \times 10^{43}$ erg/s. This limit is found consistent with a rather simple but quite complete analytical calculation of the total sensitivity of the instrument. As can be seen in Fig. 7 of Augustin et al.,[24] FIREBall-2 is presently lacking a factor of 3 in sensitivity to constrain the models at $z \sim 0.7$.

For the diffuse CGM Lyα emission, the instrument-limiting flux $\sim 7''$ away from a single object center, using one resolution element on each side of the central one, translates to a surface brightness detection limit of 1.7×10^{-16} ergs/cm^2/s/arcsec^2 (721,000 LU).

Limiting flux performance is, of course, lower than what was expected without the spurious background but can be used as a starting point for evaluating the instrument's ultimate sensitivity.

5 Ultimate Sensitivity

The operational success of the 2018 flight and clear paths toward correcting the issues that reduced the sensitivity motivated CNES and NASA to fund the project for at least one more flight.

5.1 *Major Improvements for 2021 Flight*

Table 3 describes the different sources of noise as observed in 2018 and expected in 2021. It shows that the major offender to the sensitivity is the high sky level, mostly due to the moon-scattered light, as discussed in Sec. 4.1. Because of the very short turnover wind period, launch opportunities are scarce and can occur very close to the full moon, as happened in 2018. A launch far from full moon phase will always be preferable, but stray light mitigation has to be sufficient so that even in the worst scenario, sky noise would not impact FIREBall-2's performance.[25]

Our simple SNR model shows that to achieve this goal, it is necessary to cut stray light to the UV detector by a factor larger than 200. In this case, the sky background will be less than 3 counts/PSF/h. Refinements of the stray light/off-axis reflections analysis are ongoing, but it is already clear that three baffling stages need to be set up: a gondola top cap combined with a lower baffle will prevent direct view of the balloon from the spectrograph entrance (direct view of the moon was already baffled). It will be aided by an improved system of baffles internal to

the tank. A more stringent moon-angular-distance limit and extensive testing of the baffling will ensure in-specs background in the worst expected conditions.

We experienced some "additional background noise" which we distinguish from the "detector dark noise" (Table 3). Indeed, during the flight we observed a signal, when the shutter is closed, an order of magnitude higher than the ground-based dark at the same temperature. Even though it is not totally understood, we think it is not an increased detector dark but rather a consequence of either Cerenkov emission coming mostly from the back of the FF2 mirror, a guider camera leak, or (less likely) a shutter failure that would have let ~ 1 cm^2 of light through. We have decided to reduce any possible Cerenkov contribution by masking all the mirrors' backs and are currently addressing the other possible causes. We have also added more comprehensive baffling around the immediate detector environment. It is important to stress that the level of "additional background" (shutter closed) might come from other unknown causes as it is the very first flight of these-state-of-the-art devices (large number of very low energy CRs or longer than estimated smeared CR tails).

In addition, a new detector system will be used for the next flight. It will be equipped with a new version of the detector controller (Nüvü v3), developed specifically by Nüvü Cameras for these EMCCD detectors.[14] The expected lower read noise provided by this controller combined with a slightly higher temperature should make the photon-counting algorithm more efficient. Data acquired with this new setup will allow further optimization of the temperature/smearing/gain parameters in order to increase the SNR.

The improvements in the spectrograph alignment and guidance algorithm also promise an image quality increase from $7''$ to $\sim 5 - 6\,''$, which improves our chances of disentangling CGM signal from the galaxy signal and beginning to spatially resolve CGM emission.

5.2 *2021 Flight Predicted Sensitivity*

In order to analyze the advantage of using photon counting, we compare here two scenarios for the next flight: with [Counting] or without [Analog] thresholding photon-counting process. We assume a 200-factor sky reduction and that the additional background level (as defined in Sec. 5.1) is mitigated by a factor of 10. To compute the SNR, we use one resolution element on each side of the central one for 144 exposures, amounting to 2 h observing time. The effect of CR mentioned in Sec. 3.3.1 is taken into account. It is important to remind that when stacking over a long period of time, the SNR does not increase as fast as $\sqrt{t}$. We accounted for that with the same 1.4 factor observed between 50 s and 30 min.

The noise contributions for these two cases are reported in Table 3 so that they can be compared with 2018 values.

Table 3 FIREBall-2 noise [cts/PSF/field (2 h)] contribution using one resolution element on each side of the object. Aside from the read noise, all noise analog values account for the $\sqrt{2}$ excess noise factor coming from the stochastic nature of the CCD amplification. In photon-counting mode, this factor is close to 1.

2018 Parameter Units = cts/PSF/field (2 h)	2018 flight analog mode	2021+ flight analog	2021+ flight counting	Comment
Read-out noise (50-s images)	6.4	5	0.8	Nüvü v3 and γ-counting
Clock-induced noise	8.3	6.5	4.6	Only sharper PSF
Detector dark noise	3.9	3.0	2.1	Measured in lab
Additional background noise	21.8	5.4	3.8	Shutter close, Cerenkov?
Sky noise	90	5.8	3.4	See better baffling (Sec. 4.1)
Shot noise	31.3	13.3	7.1	For a 5σ object
Total noise	$\sim$98.6	$\sim$17.8	$\sim$10.1	$\sqrt{\sum \mathrm{Noise}^2}$

In both scenarios, FIREBall-2 appears to be photon starved: the biggest offender (highest noise contribution) is the shot noise. In the photon-counting scenario, stacking 10 targets with similar galaxy properties/morphologies would decrease the shot noise to a level comparable to the other noise sources. This would make the major noise contributors (CIC, sky, background, and shot noise) very similar, prohibiting any further gain.

As expected in photon-counting mode, the read noise contribution becomes much smaller and part of the CIC noise disappears as the serial CIC is only amplified by an average gain of 200 instead of 1400 e^-/e^-. The noise was computed assuming 30% count loss and only 85% read noise reduction because of smearing when applying the photon-counting thresholding process. These values are representative of the expected improvement of the CTE with respect to the 2018 values.

The fact that, in both scenarios, the detector dark noise is not dominant might allow for a higher detector temperature, which would improve its CTE and facilitate photon counting.

For these two 2021 cases, the respective limiting 5σ total Lyα emissions are $\sim 4.3 \times 10^{41}$ and 5.3×10^{41} erg/s for a $\sim 5''$ unresolved $z \sim 0.7$ target. For the diffuse CGM Lyα emission limit and using the same definition as in Sec. 4 (with $5''$), the surface brightness detection limits are, respectively, 1.4×10^{-17} and 1.8×10^{-17} ergs/cm^2/s/arcsec^2 (61,000 LU and 75,000 LU). As we can see, these limiting fluxes for the two scenarios are very close despite very different total noise. This is due to the fact that the high photon-counting threshold used to mitigate the read noise and the serial CIC, coupled with smearing, generates event loss and therefore lowers the total efficiency by $\sim 30\%$.

In order to compare FIREBall-2's performance with that of the first generation of the instrument, we gathered in Table 4 the characteristics and performance of FIREBall-1 and 2 for their respective flights. In Table 4, the sensitivity limit is computed for 2 h acquisition on an extended $30''$ object such as a bright Quasi Stellar Object QSO.

5.3 Comparison with Existing Observations

For total flux (unresolved source), according to GALEX observations and simulations (Fig. 7 of Augustin et al.[24]), the ~ 41.7 $\log(L_{\mathrm{Ly}\alpha})$ limit we find for both scenarios in Sec. 5.2 would be

Table 4 Evolution of FIREBall scientific performances.

Parameter	FIREball-1 2009 Flight	FIREball-2 2018 Flight	FIREball-2 2021+ Flight	R^α	Rationale
Detector QE	0.07	0.55	0.55	8	MCP $\rightarrow$ EMCCD
Effective QE	0.07	0.41	0.37	5.3	CR impact, γ-counting loss
Instrument efficiency	0.0036	0.13	0.13	36	Fiber $\rightarrow$ slit, grating, misc
Atm transmission	0.4	0.4	0.5	1.25	Higher altitude
Net efficiency$^\beta$	0.0001	0.021	0.024	238	
Spatial resolution FWHM	12$''$	7$''$	5$''$	2.4	Resolve CGM regions
Total noise (cts/PSF/h)	~ 0.7	~ 67	~ 5	0.15	2021 optimistic scenario
Effective FOV (arc min^2)	8	900	900	112	Field corrector, slit mask
Wavelength range [dz]	150 Å [0.12]	75 Å [0.06]	75 Å [0.06]	0.5	Emphasize high t_{atm} band
# of CGM region/field	1	50	50	50	IFU $\rightarrow$ Multi slit mask
Selection	Blind	Preselected	Preselected		
Sensitivity (2 h, 30$''$)	74,000 LU	$\sim 80,000$ LU	$\sim 8,000$ LU	9	Direct detection (5σ)

Notes: α: Improvement ratio between 2009 FB-1 flight and 2021 FB-2 flight. β: Net effi. = Effective QE $\times$ Instr. eff $\times$ Atm trans.

constraining by an order of magnitude for the Lyα integrated flux at $z \sim 0.7$. Stacking 10 objects per flight with similar galaxy properties/morphologies, using a very accurate continuum sub-traction, should decrease this limit to $\sim$41.1.

Concerning diffuse CGM emission, observations at higher redshift with MUSE[4] revealed some Lyα haloes at $z \sim 3$ reaching $4''$ ($\sim$30 kpc) from the galaxy at their limiting surface brightness of $\sim 10^{-19}$ erg s^{-1} cm^2 arcsec^{-2}. The discussion in the work by Sullivan et al.[3] suggests that average Lyα surface brightness in the vicinity of QSOs might decrease with time (study from $z \sim 3$ to $z \sim 2$). With the assumed angular resolution of $5''$ for the 2021 flight, the first FIREBall-2 resolution element after the center is already at $\sim$35 kpc at $z \sim 0.7$. The same physical distance at $z \sim 3$ ($4.4''$) is just beyond Wisotzki et al.[4] published radius.

For a spectrally unresolved object, the favorable $\sim \times 30$ cosmological emission line surface brightness dimming at $z \sim 3$ with respect to 0.7 makes the FIREBall-2 single-object limit translate to $\sim 14.7 \times 10^{-19}$ ergs^{-1} cm^2 arcsec^{-2} at $z \sim 3$, a value five times brighter than MUSE's.[4] Stacking 25 or more galaxies[24] are thus expected to provide a detection surface brightness limit at $z \sim 0.7$ comparable to that of MUSE at $z \sim 3$ for individual galaxies. <mark>FIREBall detection of extended emission would thus only be possible in presence of large spectral offsets due to strong winds</mark>. Because of this, quasars, which have a brighter and more extended diffuse Lyα emission, are an even more interesting candidate for the 2021 flight.[2]

6 Conclusion

The 2018 flight of FIREBall-2 performed the first space multi-object acquisition using an MOS, demonstrated the successful performance of all subsystems, and provided a test flight for several key technology innovations such as the EMCCD,[14] the sub-arcsecond pointing system,[15] or the aspherical grating.[13] In this paper, we have described the calibration of the second generation of the FIREBall instrument and its performance.

FIREBall-2 gained a factor $\sim$10 in sensitivity compared to the first generation of the instrument (mostly detector and optical throughput) and now has the capability to target $\sim$50 galaxies per mask. Unfortunately, the very high sky background on the detector due to the near full moon and the punctured balloon reflecting light to a non-baffled instrument part, combined with a shortened flight, wiped out this efficiency gain. A new baffling of the instrument is being implemented to mitigate stray light impact. Aside from this, only minor changes will be performed in order to launch FIREBall-2 again to minimize risks and ensure the full benefit of the factor 10 efficiency gain from generation one.

The results of the 2018 flight combined with the ultimate sensitivity analysis validate the instrument design and its sub-systems. It demonstrates its capability to make valuable detections of the <mark>CGM and circum quasar medium</mark> and to provide critical spatiospectral information about the medium surrounding $z \sim 0.7$ galaxies and QSOs. Such information is currently missing in our search for gas dynamics in the CGM at intermediate redshifts. FIREBall-2 is the only UV instrument currently proven and able to make these necessary discoveries at intermediate redshift ($0.2 < z < 2$).

Once FIREBall-2 puts a constraint on CGM emission at $z \sim 0.7$, it motivates more detailed follow-up with larger-class space missions, as they would finally know the sensitivity, spatial, and spectral resolution constraints required to successfully map this emission. The implementation of this project is thus an investment for future orbital projects such as LUVOIR or MIDEX-class telescopes (e.g., HALO). This experiment, developed in the limited range of the stratospheric flights budget, allows both innovation with higher risk and low-cost iterative improvements.

Acknowledgments

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