

A self-deploying and self-stabilizing helical antenna for small satellites[☆]



J. Block^{*}, A. Bäger, J. Behrens, T. Delovski, L.-C. Hauer, M. Schütze,
R. Schütze, T. Sprowitz

German Aerospace Center (DLR), Germany

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ABSTRACT

Space antennas with a helical geometry are an advantageous choice for many applications, for instance if the transmission of electromagnetic waves with a circular polarization is intended, or if signals from terrestrial objects shall be received with a high angular resolution. In all these cases the desired electromagnetic properties of a helical geometry can be combined with the mechanical advantage that the antenna acts as a compression spring, provided that its core structure has the necessary high spring stiffness but can nevertheless easily be compressed. Such an antenna has been developed by DLR Institutes in Bremen and Braunschweig together with some industrial partners for a small satellite named AISat, which shall be able to pursue the position of individual ships in critical sea areas in order to improve the security of seafare trade. The development was very challenging since the antenna must expand from a stowed stack length of only 10 cm to a total length of 4 m. Only a special carbonfiber core under the conductive coating and a system of stabilizing cords led to a satisfying solution. Both the self-deployment and the self-stabilization function of this innovative antenna concept have been successfully tested and verified under zero-g-conditions in the course of a parabolic flight campaign. It could be convincingly demonstrated that the helical antenna can really achieve its desired contour in weightlessness within some seconds and maintain the required stability. Beyond the current application for the AISat satellite it is therefore quite a promising concept for future satellites.

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1. Introduction

Helical antennas offer some outstanding properties in comparison with other types of antennas [1,2]: They are ideally suited for transmitting and receiving circularly polarized electromagnetic waves, and they have an excellent directionality. Their gain increases with the number of windings, at least up to 7 or 8 windings. The higher the

achievable gain, the narrower becomes the angle of aperture. On their backside helical antennas need a conductive plate, which acts as reflector.

Due to these characteristics helical antennas are an ideal option for space applications where a satellite in orbit shall “illuminate” only a comparatively small footprint area on ground. A paradigmatic example for such a mission is the AISat project, which will be explained in the following. The innovative antenna concept that has been developed for this project could be easily transferred also to other small satellites in the future.

1.1. The AISat project

AISat is a suitcase-sized nanosatellite with a 4 m long helical antenna. The ‘AIS’ in the mission name stands for *Automatic Identification System*—a radio system designed for

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^{*} Corresponding author.

E-mail addresses: joachim.block@dlr.de (J. Block),
annette.baeger@dlr.de (A. Bäger), joerg.behrens@dlr.de (J. Behrens),
toni.delovski@dlr.de (T. Delovski), lars-christian.hauer@dlr.de
(L.-C. Hauer), martin.schuetze@schuetze-staebe.de (M. Schütze),
rainer.schuetze@schuetze-staebe.de (R. Schütze), tom.sproewitz@dlr.de (T. Sprowitz).

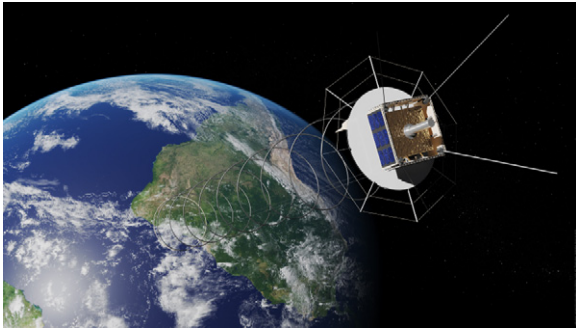


Fig. 1. AISat in space (artist's view).

the exchange of navigation, position and identification data from each and every vessel, to make global maritime traffic safer and easier to control. The excellent directionality of the helical antenna shall allow to precisely locate individual ships from an orbital position in ≈ 600 km height, even if the sea route is very busy and the distance between the ships is small, and to receive *Search And Rescue (SAR)* signals sent from survival craft or distressed vessels (Fig. 1).

Upto now this was not possible: Firstly, satellites usually receive signals from a much larger area, and secondly, many interfering signals are present in adjacent frequency bands.

1.2. The antenna concept

It is evident that a spaceborne helical antenna must be compressible and stowable in a very small volume during launch, in particular, if the relation between the antenna length (4 m) and the satellite size is so disproportionate as in this case. Consequently, the helical geometry, which was already predefined by the aforementioned electromagnetic requirements, shall be utilized for a second objective, namely to enable the antenna to act mechanically as a compression spring. Once in orbit, the spring will be released and immediately deploy itself until it has achieved its full operational length. In this final position, thin retracting cords will limit the further expansion against the residual spring force (Fig. 2).

The chronological separation of mechanical and electrical functions suggests a physical separation of the antenna materials as well: While the antenna core needs a high spring stiffness but no electrical conductivity, the properties of the outer skin must be just *vice versa*: an excellent electrical conductivity, but no mechanical stiffness at all. Therefore it was decided to cover the core with a thin metallic fabric tube consisting of tinned copper wires. The core itself is a cylindrical sandwich strut, which obtains its mechanical stiffness from a fiber fabric hose, that surrounds again an inner core made of hard foam. Preferably the fibers are carbonfibers, but glassfibers or aramide fibers could alternatively be selected (depending on the required spring constant).

1.3. AISat project partners and schedule

Two institutes of the German Aerospace Center (DLR) are involved in the AISat project: The Institute of Space Systems



Fig. 2. AISat breadboard model with deployed antenna.

in Bremen developed the main body of the nanosatellite and performed the whole system engineering, while the DLR Institute of Composite Structures and Adaptive Systems in Braunschweig took the responsibility for the experimental

verification of the helical antenna. The Schütze Company in Dorsten performed the manufacturing of the antenna hardware and for corresponding material creep tests.

As far as electrical engineering and AIS signal processing are concerned, DLR's key partners are the Bremen University of Applied Sciences and a consulting company (Joachims). However, their contributions are not discussed in the present paper.

The maiden flight of the first AISat satellite is already planned for late 2011. The satellite will be launched on a Polar Satellite Launch Vehicle from the Satish Dhawan Space Center on the Island of Sriharikota, India.

2. Design and manufacturing

The manufacturing process of the helical antenna consists actually of four consecutive processes.

2.1. Primary composition

At first, the (carbon-)fiber fabric hose is pulled upon the cylindrical hard foam core, impregnated with epoxy resin, and then surrounded with a thin PE hose, which serves as a means for radial shrinkage. This process is almost identical with the well-proven production process of straight carbonfiber struts [3], apart from two differences: The shrinkage hose consists of PE instead of glassfibers, and there are no unidirectional fibers in longitudinal direction, because they would inevitably tend to buckle on the inner side of the windings (which cannot occur on straight struts) (Fig. 3).

2.2. Building the helix

In the second step, this still very soft sandwich structure is wound upon a much larger cylindrical form, which reflects the geometry of the helical antenna (see Figs. 4 and 5). This step is the most challenging task at all, because several shrinkage and creep effects have to be taken into account in advance. In longitudinal direction, the overall length of the helix must be significantly larger during manufacturing (i.e. during the winding upon the cylindrical form) than in the nominal design case (Fig. 6). In radial direction, it must be considered that any

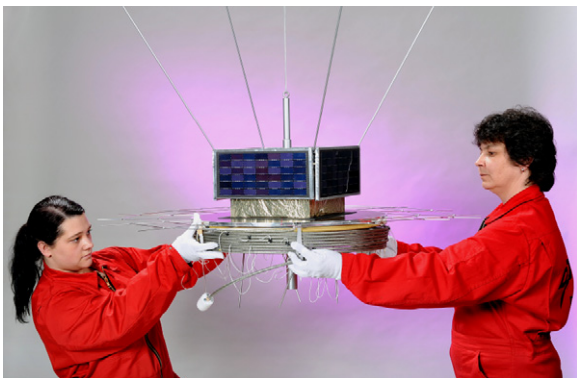


Fig. 3. AISat breadboard model with stowed antenna.

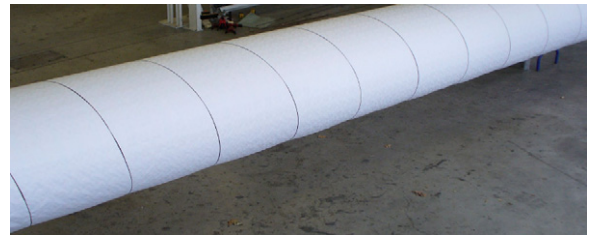


Fig. 4. Cylindrical winding form (the later helical geometry marked with a carbonfiber filament).

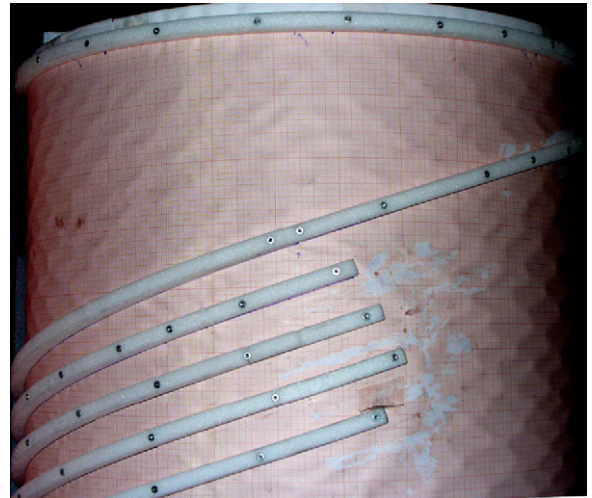


Fig. 5. Lay-up traces (foam beds) on a winding form.

elongations and shortenings of the helix will simultaneously influence the winding radius.

Once the desired manufacturing contour has been fixed upon the cylindrical form, the whole helix is being hardened and cured at elevated temperatures (+60 °C and +80 °C). But even after curing it remains subject to creep effects: Long-term stowage in the compressed configuration will significantly reduce the overall length upon sudden release, and only after longer relaxation periods the material will slowly “regenerate” (Fig. 7).

2.3. Application of the conductive coating

Finally, when the cylindrical form is removed and the curing at +80 °C is accomplished, the ready antenna core can be covered with the metallic fabric hose. At the root of the first antenna winding, however, there should remain an electric “gap”, i.e. after mounting on the reflector plate there should be no conductive connection. Instead, the metallic fabric hose will be connected to the electric circuitry on board the satellite by a suited cable.

2.4. Application of the control cords

The creep and relaxation effects described above make it inevitable to manufacture the helices with sufficient

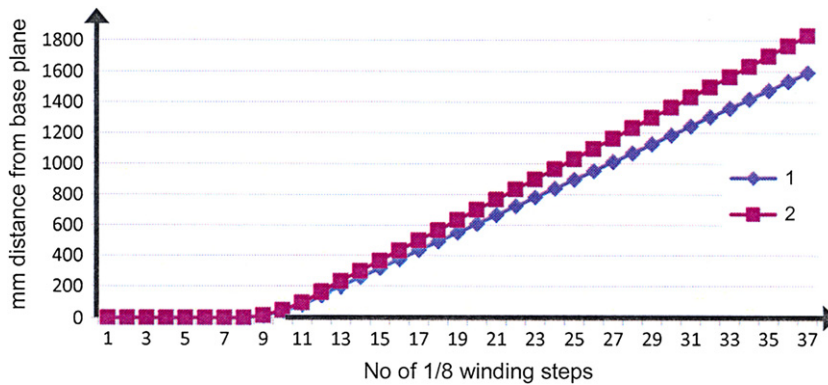


Fig. 6. Additional pitch (curve 2) must be foreseen in comparison to the nominal geometry (curve 1). The position on the cylindrical winding form is measured and fixed eight times per winding, i.e. all 45° .

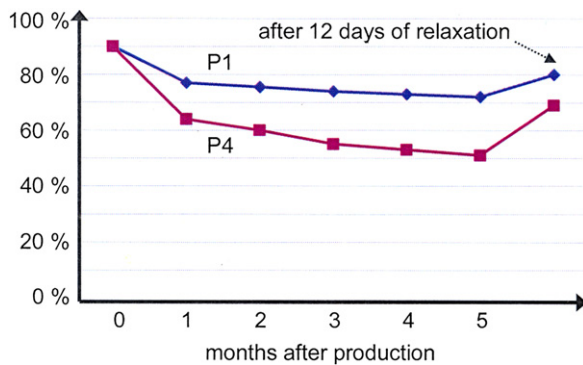


Fig. 7. Length reduction after long-term stowage in the compressed state.

“stiffness reserve”, because it must be secured that the nominal operational dimensions are achieved after deployment in space under all circumstances. Therefore the manufacturing length will have to be significantly longer than the final nominal length and that, on the other hand, the residual spring forces after deployment will be considerable, in particular when the antenna is still new. This means that in circumferential direction three electrically isolating control cords are required (with an angular distance of 120° between them), which have to counteract the residual spring force and to align the deployed antenna very accurately. They are fastened on precisely measured points on the top winding of the helix and, with their other end, on the reflector plate.

3. Test configuration for zero-G

It is obvious that a so-called “gossamer” structure as the helical antenna described above could never be satisfactorily tested under normal gravity. Apart from all other disturbing effects, the antenna would already be deflected downwards by its own weight immediately after release. On the other hand, the antenna could never be installed on AISat (or on any other satellite) without having been thoroughly tested and qualified. Therefore it became

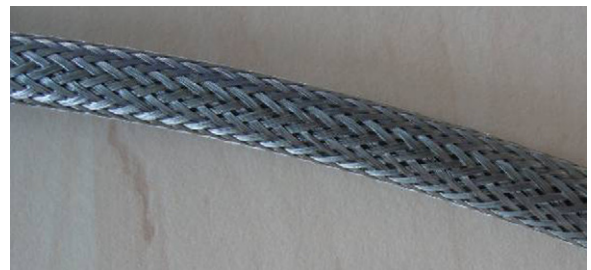


Fig. 8. Detailed view of the conductive coating. The tested antenna core diameters were 7 and 8 mm.

mandatory to test the release function, to monitor the self-deployment, and to verify the achievement of a well-aligned final attitude in space within a parabolic flight where up to 22 s of weightlessness per parabola are available.

Consequently, four slightly different types of the AISat antenna were tested in the course of the 15th DLR Parabolic Flight Campaign in Bordeaux (France) on March 9, 2010 (Fig. 8).

These antennas were stored in their stowed configuration in a magazine on the test rack. For each test series (5 parabolas) one of them was taken out and locked on a circular antenna ground plate (a sandwich plate) in vertical position (Fig. 9). The test rack itself was a heritage from a previous DLR test campaign [4].

Once released, the antennas should safely unfold up to a total length of ≈ 4 m, with a helix diameter of 0.57 m. The three release mechanisms are manually triggered upon injection into the zero-g phase by cutting a circumferential string, which holds their hold-down flanges down (Fig. 10). They did not need any control electronics. In space, one could use a simple cutter.

The antennas were mounted on the triangular antenna ground plate via force transducers so that the reaction forces upon deployment could be measured.

Simultaneously, three video cameras fastened on the test rack and two additional cameras mounted on the aircraft handrail behind the test rack, ≈ 4 m ahead of it, monitored the unfolding process. All cameras were

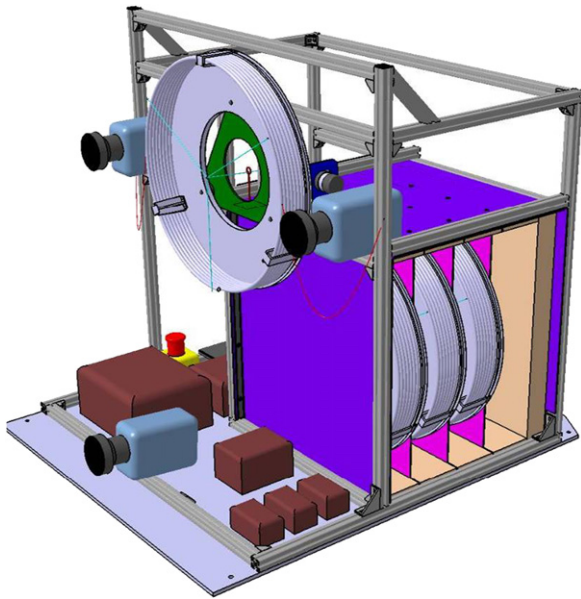


Fig. 9. Test rack for four antennas with identical geometry, but different material combinations in the helix core (carbonfibers, glassfibers and aramide fibers).

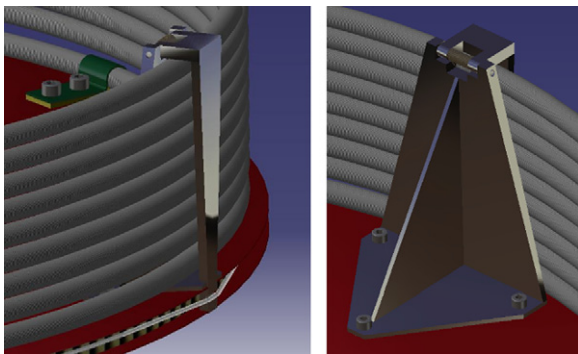


Fig. 10. Outside and inside view of one of the release mechanisms. The hinge is driven by a torsion spring.

self-recording. They were switched on prior the first parabola and switched off after the last parabola.

In addition to the control cords discussed above there were some additional glassfiber cords for safety and retraction purposes, due to the strict safety regulations on board the aircraft [5] (Fig. 11).

4. Results

4.1. Experimental observations

The 31 parabolas performed during the flight were divided into 6 series of 5 parabolas each, plus one initial parabola to become familiar with the procedure. Table 1 summarizes the test plan for the AISat antenna. Fig. 12 shows some typical snapshot photos (Fig. 13).

It could be successfully verified that within the 22 s of zero-g conditions in each parabola the helical antenna

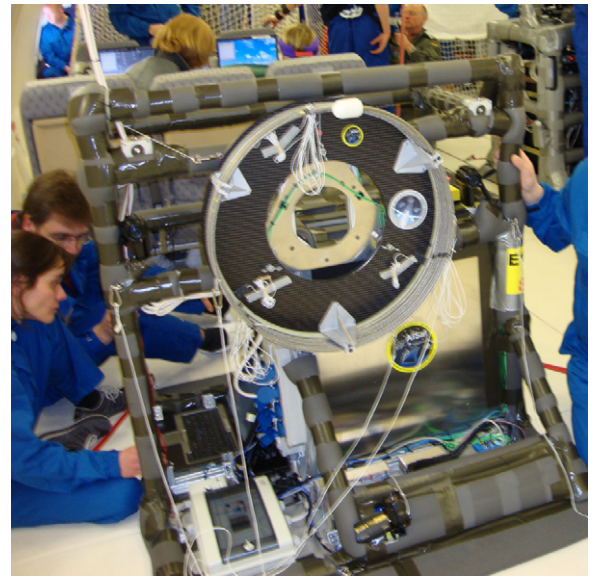


Fig. 11. Test rack with stowed antenna. The white loops hanging down from the antenna are the control cords.

Table 1

Test plan for the AISat antenna.

Parabola sequence	Test object	Composition	String diameter
1	Antenna 1	CFRP+GFRP	9.5 mm
2	Antenna 2	Aramide FRP	9.5 mm
3	Antenna 3	CFRP+GFRP	8.5 mm
4	Antenna 4	Aramide FRP	8.5 mm
5	Repetition of most promising sequence (1)		
6	as 5 and photographic documentation		

structure can be completely deployed to its full length. This is the most decisive result, which proves the feasibility of the concept. However, the presence of the control cords leads to strong vibrations induced by the rebounds, which occur when the cords are tightened.

As far as the directionality of the forward motion is concerned, it was observed that friction of the antenna windings on the flanges of the release mechanisms has a strong effect on the initial direction into which the unfolding antenna moves. Here a design improvement became mandatory (and was realized very soon after the test campaign). If the initial expansion is strongly misaligned then the points in time when the three control cords reach their final length will strongly differ, so that the antenna rebounds in an asynchronous manner, and the vibrations become worse.

Finally it became obvious during the test campaign that high material stiffness is advantageous, so that the final AISat antenna (the flight unit) will have a carbonfiber-dominated core (CFRP) with high modulus.

4.2. Comparison with numerical simulations

Of course the kinematics of the antenna deployment had been simulated in advance by suited numerical tools



Fig. 12. Grasping the deployed antenna (4 scenarios).

(see Figs. 14 and 15). However, a number of almost unpredictable parameter values (such as the friction coefficient on the flanges of the release mechanisms,

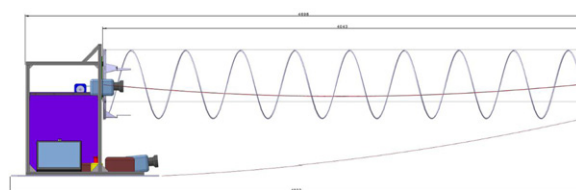


Fig. 13. Test scheme for the deployed configuration.

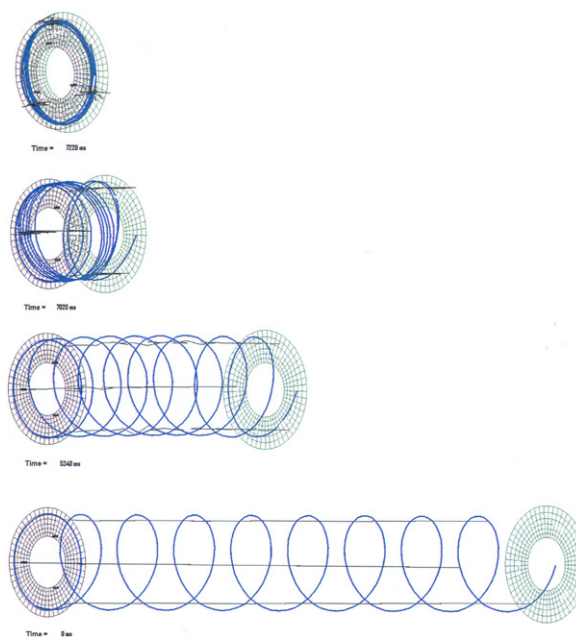


Fig. 14. Simulated unfolding, ideally straight.

alignment imperfections or small inhomogeneities in the stacking geometry) have an influence, which makes it practically impossible to accurately predict the movement of the individual antenna. Already very slight variations of those parameters result in a lot of different individual unfolding scenarios, which are nevertheless characteristic in general.

Indeed the numerical simulations performed on the computer showed the same typical kinematic performance as the real test runs on the zero-g test flight. In particular, they finally resulted in the same end configuration, namely the fully deployed antenna, converging to an equilibrium between residual spring force (expansion) and tension in the control cords (retraction). It was therefore decided to regard the results as representative and to accept them as flight qualification for the AISat project.

5. Conclusions

The most decisive lessons learned during the antenna manufacturing, and the experimental results obtained

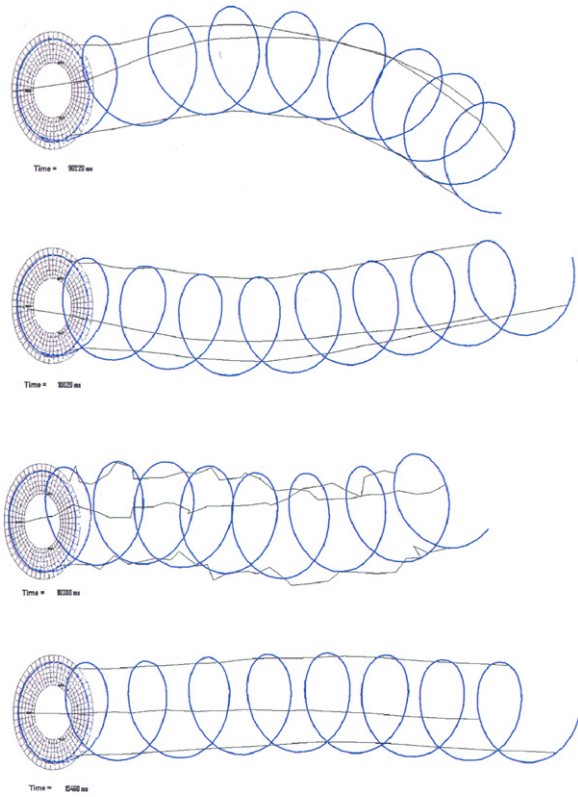


Fig. 15. Simulated unfolding with intermediate lateral deflections. The amplitude of the deflections decreases steadily.

during the zero-g test campaign, can be summarized as follows:

- The concept of a helical antenna that can be stowed and then, upon release, expand like a compression spring, is feasible.

- With a stiffness reserve that is large enough to compensate losses induced by material creep and relaxation, and with suited control cords, which limit the expansion, the helices will achieve a stable final length that can be adjusted in accordance with the electromagnetic requirements.
- The direction into which the antenna moves forward immediately after release depends strongly on the uniformity of its interaction with the release mechanisms. High-quality release mechanisms with little friction will help to make the expansion as straight as possible.
- The vibrations induced by the expansion and, in particular, by rebounds of the antenna tip when the control cords are tightened, decrease obviously; although there was not sufficient time during the zero-g phases to make the antenna calm.
- The overall performance of carbonfiber-based antenna cores is superior to competing types. Consequently the AISat flight model will be equipped with such an antenna.

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