

$$1, \quad \ddot{u} = \frac{\Sigma - mg \sin \theta - mgw + mr\dot{v}}{m}$$

$$u_0 = v_0 = w_0 = 0 \quad L_0 = M_0 = N_0 = 0$$

$$p_0 = q_0 = r_0 = 0$$

$$\psi_0 = \phi_0 = \theta_0 = 0$$

$$\Sigma_0 = \Sigma_0 = \Sigma_0 = 0$$

$$h_1 \quad \ddot{u} = \left(\frac{\Sigma}{m} - g \sin \theta - g w + r \dot{v} \right)$$

$$\Delta \ddot{u} = \frac{\partial h_1}{\partial \Sigma} \Big|_0 \Delta \Sigma + \frac{\partial h_1}{\partial g} \Big|_0 \Delta g + \frac{\partial h_1}{\partial w} \Big|_0 \Delta w + \frac{\partial h_1}{\partial r} \Big|_0 \Delta r + \frac{\partial h_1}{\partial v} \Big|_0 \Delta v + \frac{\partial h_1}{\partial \theta} \Big|_0 \Delta \theta$$

$$\Delta \ddot{u} = \frac{\Delta \Sigma}{m} + g \cos \theta \Delta \theta - u_0 \Delta g - g_0 \Delta w + r_0 \Delta r + v_0 \Delta v$$

$$\Delta \ddot{u} = \frac{\Delta \Sigma}{m} + g \Delta \theta \quad (1)$$

$$\ddot{v} = \frac{\Sigma + mg \cos \theta \sin \phi - mr u + mp w}{m}$$

$$\dot{v} = \frac{\Sigma}{m} + g \cos \theta \sin \phi - r u + p w$$

$$\dot{v} = \dot{v}_0 + \Delta \dot{v}$$

$$= h_2 ()$$

$$\Delta \dot{v} = \frac{\partial h_2}{\partial \Sigma} \Big|_0 \Delta \Sigma + \frac{\partial h_2}{\partial \theta} \Big|_0 \Delta \theta + \frac{\partial h_2}{\partial \phi} \Big|_0 \Delta \phi + \frac{\partial h_2}{\partial r} \Big|_0 \Delta r + \frac{\partial h_2}{\partial u} \Big|_0 \Delta u + \frac{\partial h_2}{\partial p} \Big|_0 \Delta p + \frac{\partial h_2}{\partial w} \Big|_0 \Delta w$$

$$\Delta \dot{v} = \frac{\Delta \Sigma}{m} - g \sin \theta \sin \phi_0 \Delta \theta + g \cos \theta \cos \phi_0 \Delta \phi - u_0 \Delta r - p_0 \Delta u + w_0 \Delta p + p_0 \Delta w$$

$$\Delta \dot{v} = \frac{\Delta \Sigma}{m} + g \Delta \phi \quad (2)$$

$$\dot{w} = \frac{\Sigma + mg \cos \theta \cos \phi - mp v + mq u}{m}$$

$$\dot{w} = \frac{\Sigma}{m} + g \cos \theta \cos \phi - p v + q u$$

$$= h_3 ()$$

$$\Delta \dot{w} = \frac{\partial h_3}{\partial \Sigma} \Big|_0 \Delta \Sigma + \frac{\partial h_3}{\partial \theta} \Big|_0 \Delta \theta + \frac{\partial h_3}{\partial \phi} \Big|_0 \Delta \phi + \frac{\partial h_3}{\partial p} \Big|_0 \Delta p + \frac{\partial h_3}{\partial v} \Big|_0 \Delta v + \frac{\partial h_3}{\partial q} \Big|_0 \Delta q + \frac{\partial h_3}{\partial u} \Big|_0 \Delta u$$

$$\Delta \dot{w} = \frac{\Delta \Sigma}{m} + g \sin \theta \cos \phi_0 \Delta \theta - g \cos \theta \sin \phi_0 \Delta \phi - v_0 \Delta p - p_0 \Delta v + q_0 \Delta u + q_0 \Delta u$$

$$\Delta \dot{w} = \frac{\Delta \Sigma}{m} \quad (3)$$

$$X = -\eta \sqrt{u^2 + v^2 + w^2} \cdot u$$

$$= g_1(x)$$

$$\Delta X = \frac{\partial g_1}{\partial u} \Delta u + \frac{\partial g_1}{\partial v} \Delta v + \frac{\partial g_1}{\partial w} \Delta w$$

$$\Delta X = -\eta \frac{2u_0^2 + v_0^2 + w_0^2}{\sqrt{u_0^2 + v_0^2 + w_0^2}} \Delta u - \eta \frac{u_0 v_0}{\sqrt{u_0^2 + v_0^2 + w_0^2}} \Delta v - \eta \frac{u_0 w_0}{\sqrt{u_0^2 + v_0^2 + w_0^2}} \Delta w = 0$$

$$(1) \boxed{\Delta \dot{u} = g \Delta \phi}$$

$$Y = -\eta \sqrt{u^2 + v^2 + w^2} \cdot v$$

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$$\Delta Y = 0$$

$$(2) \boxed{\Delta \dot{v} = g \Delta \phi}$$

$$Z = \eta \sqrt{u^2 + v^2 + w^2} \cdot w + Z_c$$

$$= g_3(x)$$

$$\Delta Z = \frac{\partial g_3}{\partial u} \Delta u + \frac{\partial g_3}{\partial v} \Delta v + \frac{\partial g_3}{\partial w} \Delta w + \frac{\partial g_3}{\partial f_1} \Delta f_1 + \dots + \frac{\partial g_3}{\partial f_4} \Delta f_4$$

$$\Delta Z = -\eta \frac{2u_0^2 + v_0^2 + w_0^2}{\sqrt{u_0^2 + v_0^2 + w_0^2}} \Delta u - \eta \frac{u_0 v_0}{\sqrt{u_0^2 + v_0^2 + w_0^2}} \Delta v - \eta \frac{u_0 w_0}{\sqrt{u_0^2 + v_0^2 + w_0^2}} \Delta w + \Delta f_1 + \Delta f_2 + \Delta f_3 + \Delta f_4$$

$$\Delta Z = \Delta Z_c$$

$$(3) \boxed{\Delta \dot{w} = \frac{\Delta Z_c}{m}}$$

$$\dot{P} = \frac{L + I_x \dot{r} + q r (I_z - I_y) - I_{zx} p q}{I_x}$$

$$I_{zx} = 0$$

$$\dot{P} = \frac{L + q r (I_z - I_y)}{I_x} = h_4$$

$$\Delta \dot{P} = \frac{\partial h_4}{\partial r} \Delta r + \frac{\partial h_4}{\partial L} \Delta L + \frac{\partial h_4}{\partial q} \Delta q$$

$$\Delta \dot{P} = \frac{q_0 \Delta r (I_z - I_y)}{I_x} + \frac{\Delta L}{I_x} + \frac{\cancel{0} \Delta q (I_z - I_y)}{I_x} = \frac{\Delta L}{I_x} \quad \Delta \dot{P} = \frac{\Delta L}{I_x} \quad (4)$$

$$L = -\alpha \sqrt{p^2 + q^2 + r^2} \cdot p + L_c = q_4 \quad \Delta L = \frac{\partial q_4}{\partial p} \Delta p + \frac{\partial q_4}{\partial q} \Delta q + \frac{\partial q_4}{\partial r} \Delta r + \frac{\partial q_4}{\partial L_c} \Delta L_c$$

$$\Delta L = -\alpha \frac{2 p r^2 + q_0^2 r_0^2}{\sqrt{p^2 + q_0^2 + r_0^2}} \Delta p - \alpha \frac{p q_0}{\sqrt{p^2 + q_0^2 + r_0^2}} \Delta q - \alpha \frac{p_0 q}{\sqrt{p^2 + q_0^2 + r_0^2}} \Delta r + \Delta L_c$$

$$\Delta L_c = \frac{r}{\sqrt{2}} (\Delta f_1 + \Delta f_2 - \Delta f_3 - \Delta f_4)$$

$$\Delta L = \Delta L_c$$

$$\boxed{\Delta \dot{P} = \frac{\Delta L_c}{I_x}} \quad (4)$$

$$\dot{q} = \frac{M - r p (I_x - I_z) + I_{zx} (p^2 - r^2)}{I_y} = \frac{M}{I_y} - \frac{r p (I_x - I_z)}{I_y} = h_5$$

$$\Delta \dot{q} = \frac{\partial h_5}{\partial r} \Delta r + \frac{\partial h_5}{\partial p} \Delta p + \frac{\partial h_5}{\partial M} \Delta M = -\Delta r p \frac{(I_x - I_z)}{I_y} - p \Delta r \frac{(I_x - I_z)}{I_y} + \frac{\Delta M}{I_y}$$

$$\Delta \dot{q} = \frac{\Delta M}{I_y} \quad (5)$$

$$M = -\alpha \sqrt{p^2 + q^2 + r^2} \cdot q + M_c = q_5$$

$$\Delta M = \frac{\partial q_5}{\partial p} \Delta p + \frac{\partial q_5}{\partial q} \Delta q + \frac{\partial q_5}{\partial r} \Delta r + \frac{\partial q_5}{\partial M_c} \Delta M_c$$

$$\Delta M = -\alpha \frac{p_0 q}{\sqrt{p^2 + q_0^2 + r_0^2}} \Delta p - \alpha \frac{p^2 + 2 q_0^2 r_0^2}{\sqrt{p^2 + q_0^2 + r_0^2}} \Delta q - \alpha \frac{q r_0}{\sqrt{p^2 + q_0^2 + r_0^2}} \Delta r + \Delta M_c$$

$$\Delta M = \Delta M_c$$

$$\boxed{\Delta \dot{q} = \frac{\Delta M_c}{I_y}} \quad (5)$$

$$\dot{r} = \frac{N + I_2 \cancel{p} + p q (I_y - I_x) + I_x \cancel{q} r}{I_z} = \frac{N}{I_z} - \frac{p q (I_y - I_x)}{I_z} = h_c \quad (5)$$

$$\Delta \dot{r} = \frac{\partial h_c}{\partial p} \Big|_0 \Delta p + \frac{\partial h_c}{\partial q} \Big|_0 \Delta q + \frac{\partial h_c}{\partial N} \Big|_0 \Delta N = - \frac{p_0 q_0 (I_y - I_x)}{I_z} - \frac{p_0 \Delta q (I_y - I_x)}{I_z} + \frac{\Delta N}{I_z}$$

$$\Delta \dot{r} = \frac{\Delta N}{I_z} \quad (6)$$

$$N = -\alpha \sqrt{p^2 + q^2 + r^2} r + N_c = h_c \quad (7)$$

$$\Delta N = \frac{\partial h_c}{\partial p} \Big|_0 \Delta p + \frac{\partial h_c}{\partial q} \Big|_0 \Delta q + \frac{\partial h_c}{\partial r} \Big|_0 \Delta r + \frac{\partial h_c}{\partial N_c} \Big|_0 \Delta N_c$$

$$\Delta N = -\alpha \frac{p_0 r_0}{\sqrt{p_0^2 + q_0^2 + r_0^2}} \Delta p - \alpha \frac{q_0 r_0}{\sqrt{p_0^2 + q_0^2 + r_0^2}} \Delta q - \alpha \frac{p_0^2 + q_0^2 + 2r_0^2}{\sqrt{p_0^2 + q_0^2 + r_0^2}} \Delta r + \Delta N_c$$

$$\Delta N = \Delta N_c$$

$$\boxed{\Delta \dot{r} = \frac{\Delta N_c}{I_z}} \quad (8)$$

$$\dot{\theta} = q \cos \phi - r \sin \phi = h_7 \quad (9)$$

$$\Delta \dot{\theta} = \frac{\partial h_7}{\partial q} \Big|_0 \Delta q + \frac{\partial h_7}{\partial \phi} \Big|_0 \Delta \phi + \frac{\partial h_7}{\partial r} \Big|_0 \Delta r = \cos \phi_0 \Delta q + (q_0 \sin \phi_0 - r_0 \cos \phi_0) \Delta \phi - \sin \phi_0 \Delta r$$

$$\boxed{\Delta \dot{\theta} = \Delta q} \quad (10)$$

$$\dot{\phi} = p + (q \sin \phi + r \cos \phi) \tan \theta = h_8 \quad (11)$$

$$\Delta \dot{\phi} = \frac{\partial h_8}{\partial p} \Big|_0 \Delta p + \frac{\partial h_8}{\partial q} \Big|_0 \Delta q + \frac{\partial h_8}{\partial r} \Big|_0 \Delta r + \frac{\partial h_8}{\partial \phi} \Big|_0 \Delta \phi + \frac{\partial h_8}{\partial \theta} \Big|_0 \Delta \theta$$

$$\Delta \dot{\phi} = \Delta p + \sin \phi_0 \tan \theta_0 \Delta q + \cos \phi_0 \tan \theta_0 \Delta r + (q_0 \cos \phi_0 - r_0 \sin \phi_0) \tan \theta_0 \Delta \phi + (q_0 \sin \phi_0 + r_0 \cos \phi_0) \sec^2 \theta_0 \Delta \theta$$

$$\boxed{\Delta \dot{\phi} = \Delta p} \quad (12)$$

2. The results make some sense since hover is not a stable flight condition. Any small deviation in the flight conditions cause the aircraft to leave hover.
3. The linearized version seems very similar to the non-linearized results. There is a noticeable difference in the rate changes since the rates in the linearized model don't have drag.
4. The control law makes the quadcopter semi stable since it still deviates a little over time.
5. The control law doesn't do much in terms of stabilizing the quadcopter. The copter starts oscillating for a few seconds before crashing.