

$$z) du - \frac{dp}{\rho a} = 0 \Rightarrow \frac{u_{\text{wave}}}{a_\infty} = \left(\frac{\gamma+1}{\gamma-1} \left(\frac{P}{P_\infty} \right)^{\frac{(\gamma-1)}{2\gamma}} - \frac{2}{\gamma-1} \right) + \frac{u_\infty}{a_\infty}$$

VLR wave compatibility \Rightarrow we have a right running wave meeting a region where the flow is uniform $\frac{u_{\text{wave}}}{a_\infty}$

Need to relate ρ and dp then integrate \Rightarrow assume CPI_4 ?

Yes, assume CPI_4

$$\tilde{a} = \frac{\partial p}{\partial \rho}|_S \quad \rho/\rho^\gamma = C_1 \text{ for isentropic } (S=\text{const})$$

$$a^2 = \left(\frac{\partial p}{\partial \rho}|_S \right) = \gamma C_1 \rho^{\gamma-1}$$

$$a = \sqrt{\gamma C_1} \rho^{(\gamma-1)/2}$$

$$da = \sqrt{\gamma C_1} \sum_{\gamma-1} \rho^{\frac{(\gamma-1)}{2}-1} d\rho$$

$$\Rightarrow da = \frac{\gamma-1}{2} \frac{a}{\rho} d\rho$$

$$\frac{dp}{\rho a} = \frac{a \frac{d\rho}{\rho a}}{\rho a} \quad d\rho = \frac{2}{\gamma-1} \frac{\rho}{a} da$$

$$= \frac{\tilde{a}^2}{\rho a} \cdot \frac{2}{\gamma-1} \cdot \frac{\rho}{a} da$$

$$\Rightarrow du - \frac{2}{\gamma-1} da = 0$$

integrate \Rightarrow

$$(u - u_\infty) - \frac{2}{\gamma-1} (a - a_\infty) = 0$$

$$u - \frac{2a}{\gamma-1} = u_\infty - \frac{2a_\infty}{\gamma-1}$$

$$u = u_\infty + \frac{2}{\gamma-1} (a - a_\infty)$$

$u_{\text{wave}} = u + a$ Because C+ characteristics come from uniform region

$$u_{\text{wave}} = u_\infty + a \left(\frac{\gamma}{\gamma-1} - 1 \right) - \frac{2}{\gamma-1} a_\infty$$

$$u_{\text{wave}} = u_\infty + \frac{\gamma+1}{\gamma-1} a - \frac{2}{\gamma-1} a_\infty$$

$$\frac{u_{\text{wave}}}{a_\infty} = \left(\frac{\gamma+1}{\gamma-1} \frac{a}{a_\infty} - \frac{2}{\gamma-1} \right) + \frac{u_\infty}{a_\infty}$$

$$P = Ca$$

$$\frac{P}{P_\infty} = \left(\frac{a}{a_\infty} \right)^{\frac{2\gamma}{\gamma-1}}$$

$$\Rightarrow \frac{u_{\text{wave}}}{a_\infty} = \left[\left(\frac{P}{P_\infty} \right)^{\frac{\gamma-1}{2\gamma}} \cdot \frac{\gamma+1}{\gamma-1} - \frac{2}{\gamma-1} \right] + \frac{u_\infty}{a_\infty}$$

All "particle velocity in same direction as wave propagation" means is right-running wave

$$u_{\text{wave}} = u_\infty + \frac{\gamma+1}{\gamma-1} a - \frac{2}{\gamma-1} a_\infty$$

$$\frac{\partial u_{\text{wave}}}{\partial P} = \frac{\gamma+1}{\gamma-1} \frac{\partial a}{\partial P} \quad \frac{\partial P}{\partial a} = \frac{2}{\gamma-1} \frac{\partial a}{\partial P} \Rightarrow \frac{\partial a}{\partial P} - \frac{1}{\gamma_P} > 0$$

$$\frac{\partial u_{\text{wave}}}{\partial P} = C \underbrace{\frac{\partial a}{\partial P}}_{\text{and } \frac{\partial a}{\partial P} > 0 \text{ meaning}} \quad \text{↗}$$

u_{wave} increases with increasing pressure.

In $x-t$ wave diagram, $\partial t/\partial x$ gets less steep because $\partial t/\partial x = \frac{1}{u+a}$

$$3) \quad p_1 = 58.8 \cdot 10^3 \text{ Pa} \quad T_1 = 450 \text{ K} \quad T_4 = 450 \text{ K}$$

a. Before the shock passes flow at pitot is quiescent.
Stationary

$$p_{0,1} = 58.8 \cdot 10^3 \text{ Pa} = p_{\text{static},1}$$

b. $p_2/p_1 = 3.508$

$$p_2 = p_{\text{static}} = 206.2704$$

$$\gamma = 1.4 \quad R = 287$$

$$a_1 = (\gamma R T_1)^{1/2} = 425.72 \text{ m/s}$$

$$\frac{p_2}{p_1} = 1 + \frac{\gamma - 1}{\gamma + 1} (M_1^2 - 1)$$

$$M_1 = 1.775$$

$$W_s = 1.775 \cdot a_1 = 752.66 \text{ m/s}$$

$$\frac{J^2}{p_1} = \frac{(r+1)M_1^2}{(r-1)M_1^2 + 2} = 2.32$$

$$\frac{T_2}{T_1} = \left(\frac{p_2}{p_1} \right) / \left(\frac{J^2}{p_1} \right) = 1.512$$

$$T_2 = 680.3 \text{ K}$$

$$a_2 = \sqrt{\gamma R T_2} = 522.8 \text{ m/s}$$

$$u_2 = W_s (1 - J^2/p_2) = 429.2 \text{ m/s}$$

$$M_2 = 429.2 / 522.8 = 0.821$$

$$p_{0,2} = p_2 \left(1 + \frac{r-1}{2} M_2^2 \right)^{(r+1)/(r-1)} = 321 \cdot 10^3 \text{ Pa}$$

c. Contact surface has properties 3 and pressure, velocity equal to 2

$$\frac{p_4}{p_2} = \frac{p_4}{p_3} = \left(1 - \frac{\gamma-1}{2} \frac{u_2^2}{a_1}\right)^{-\frac{2\gamma}{\gamma-1}}$$

solves to $\frac{p_4}{p_3} = 4.85$

$$T_3 = \bar{T}_4 \left(\frac{p_3}{p_4} \right)^{\frac{\gamma-1}{\gamma}}$$

$$a_3 = \sqrt{\gamma R T_3}$$

$$M_3 = \frac{u_3}{a_3}$$

$$\frac{p_{0,\text{pilot}}}{p_3} = \left(\frac{\gamma+1}{2} M_3^2 \right)^{\frac{\gamma}{\gamma-1}} \left(\frac{\gamma+1}{2\gamma M_3^2 - (\gamma-1)} \right)^{1/\gamma-1}$$

Normal shock then isentropic to stagnation

$$\underline{p_3 = 206.27 \cdot 10^3 p_1}$$

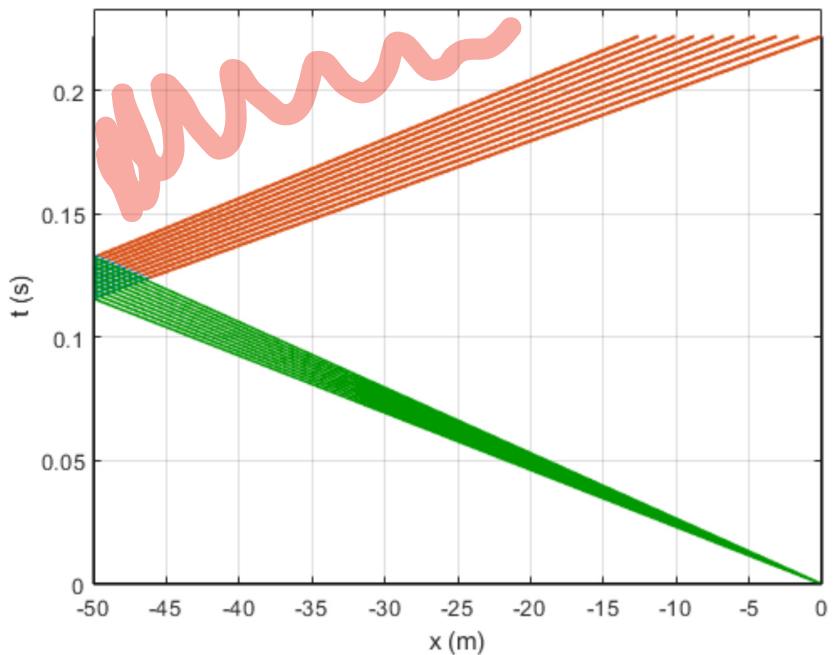
$$\underline{p_{0,\text{pilot}} = 536.8 \cdot 10^3 p_0}$$

d. $t_{\text{shock}} = \frac{3}{2a_1 c_s} = 3.98 \cdot 10^{-3} \text{ s}$

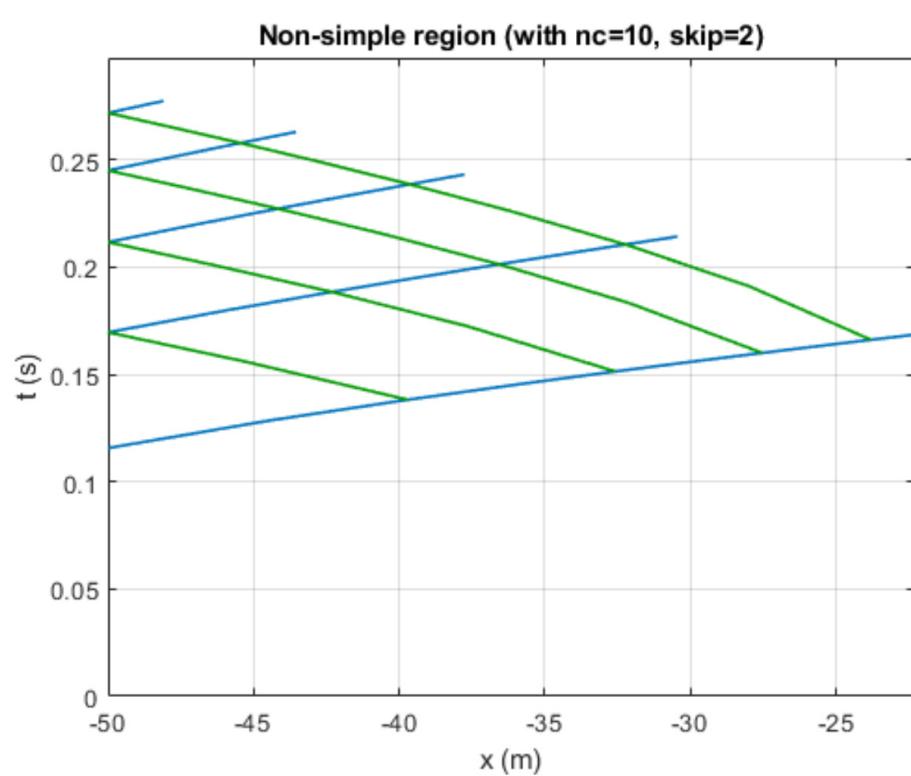
e. $\Delta t = 3(\sqrt{u_3} - \sqrt{w_3}) = 3.01 \cdot 10^{-3} \text{ s}$

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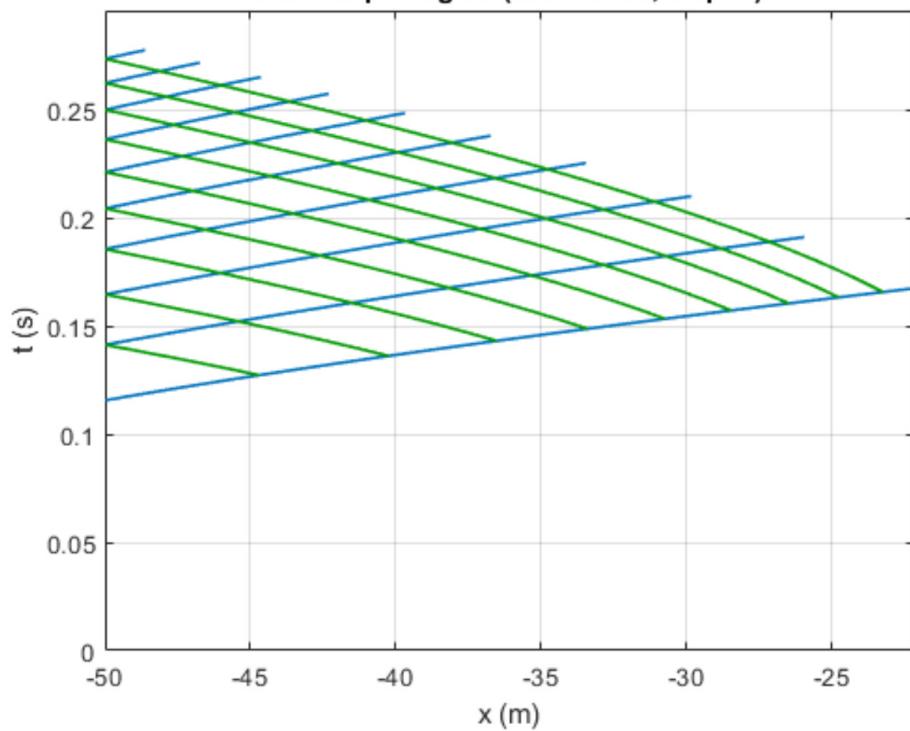
Fact 1



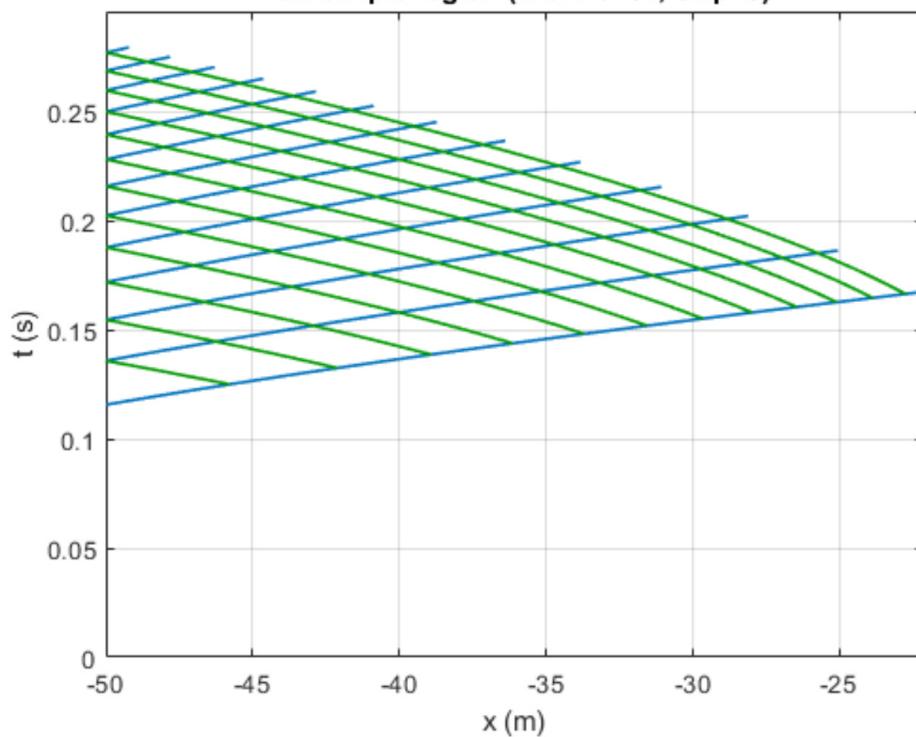
Red highlighted region has constant properties
and it makes up 72% of the tube at this time

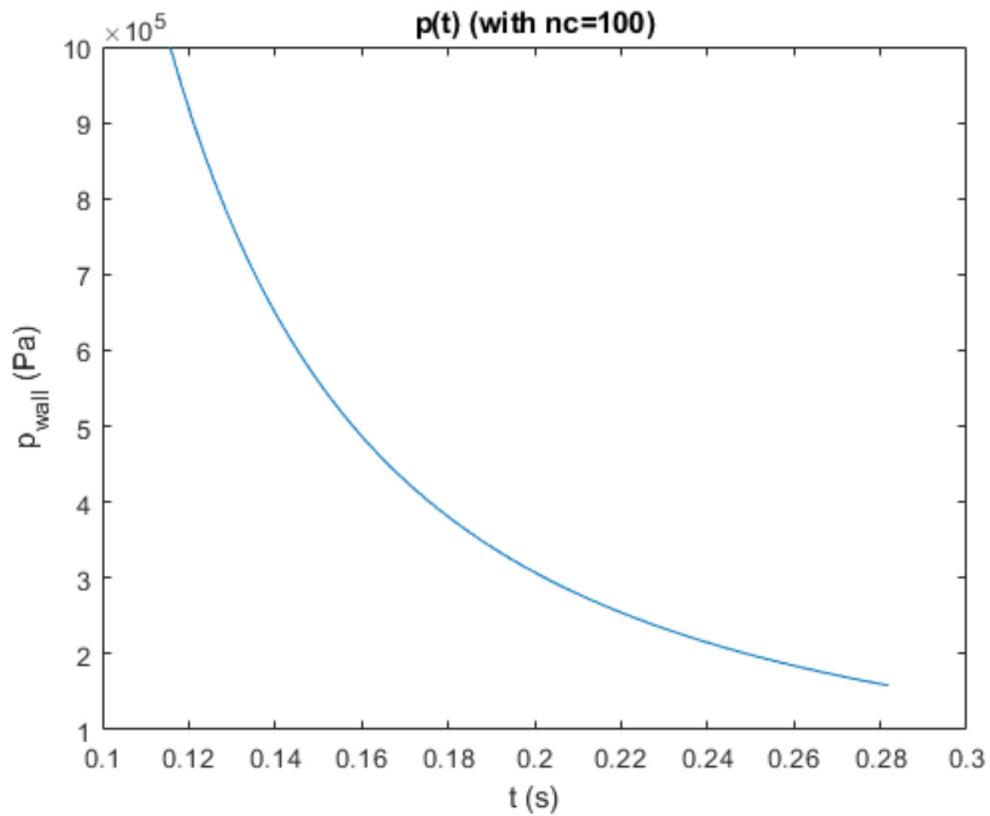


Non-simple region (with nc=50, skip=5)



Non-simple region (with nc=64, skip=5)





Code says $N=64$ is sufficient for convergence
using tail-only criterion (comparing tail-shape)