

#Problem 1

```
function FlightPathAnglesFromState(aircraft_state)

    euler_angles = EulerAngles(aircraft_state[4:6])
    wind_angles = AirRelativeVelocityVectorToWindAngles(aircraft_state[7:9])
    Vg = TransformFromBodyToInertial(aircraft_state[7:9], euler_angles)
    #=
    We know  $\gamma_a = \gamma$  when there is no wind.
    Thus,  $\gamma = \gamma_a = \theta - \alpha$ 
    =#
    #  $\gamma = \text{aircraft\_state}.\theta - \text{wind\_angles}.\alpha$ 
     $\gamma = \text{atan}(Vg[3], \text{sqrt}(Vg[1]^2 + Vg[2]^2))$ 
     $\chi = \text{atan}(Vg[2], Vg[1])$ 

    return (Vg,  $\chi$ ,  $\gamma$ )
end

# =
filename = "ttwistor.mat"
aircraft_parameters = AircraftParameters(filename)

trim_definition = TrimDefinitionCT(18.0, 0.0, 1655, 500.0)
state, control, results = GetTrimConditions(trim_definition, aircraft_parameters)
trim_variables = TrimVariablesCT(results.minimizer)
Vg,  $\chi$ ,  $\gamma$  = FlightPathAnglesFromState(state)
=#
```

Problem 4





