

NAP: A Computer Program for the Computation of Two-Dimensional, Time-Dependent, Inviscid Nozzle Flow

by

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LASL Logo
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ABSTRACT

A computer program, NAP, is presented for calculating inviscid, steady, and unsteady flow in two-dimensional and axisymmetric nozzles. Interior mesh points are computed using the MacCormack finite-difference scheme, while a characteristic scheme is used to calculate the boundary mesh points. An explicit artificial viscosity term is included for shock computations. The fluid is assumed to be a perfect gas. This method was used to compute the steady flow in a 45° - 15° conical, converging-diverging nozzle, a 15° conical, converging nozzle, and a 10° conical, plug nozzle. The numerical solution agreed well with the experimental data. In contrast to previous time-dependent methods for calculating steady flows, the computational times were < 1 min on a CDC 6600 computer.

Chapter 1

Basic Description of the Method

1.1 Introduction

The equations of motion governing steady, inviscid flow are of a mixed type: hyperbolic in the supersonic region and elliptic in the subsonic region. These mathematical difficulties may be removed by using the “time-dependent” method, in which the flow is assumed to be unsteady or time-dependent. Then the governing equations are hyperbolic in both subsonic and supersonic regions. The steady-state solution may be obtained as the asymptotic solution for large time. This time-dependent technique has been used to compute steady converging-diverging nozzle flows (reported in Refs. 1–6), and it has also been used to compute steady converging nozzle flows (see Refs. 4 and 7). The results of those calculations are mainly good, but the computational times are rather large. In addition, although the computer program of Ref. 6 included a centerbody and those of Refs. 4 and 7 included the exhaust jet, none of the above codes is able to calculate both, that is, plug nozzles.

The object of this research was to develop a production-type computer program capable of solving steady converging, converging-diverging, and plug two-dimensional nozzle flows in computational times of < 1 min on a CDC 6600 computer. Such a program would be able to solve unsteady flows as well.

1.2 Literature Review

The following is a discussion of the methods used in References 1 through 7. The first paragraph deals with the computation of the interior mesh points; the next three paragraphs are concerned with the boundary mesh points.

Prozan (see Ref. 1), Wehofer and Moger, and Laval used variations of the two-step Lax-Wendroff scheme to compute the interior mesh points. Migdal *et al.* and Brown and Ozcan employed the original one-step Lax-Wendroff scheme, but with the equations of motion in nonconservation form. Serra applied the original Lax-Wendroff scheme with the equations of motion in conservation form. To stabilize their schemes, Laval and Serra used artificial viscosity terms in their difference equations.

Wehofer and Moger reset the stagnation conditions along each streamline, reset the mass flow at each axial location, and smoothed the subsonic portion of the flow after each time step.

To compute the nozzle inlet mesh points, Prozan (in Ref. 1) assumed the inlet flow to be uniform. Wehofer and Moger assumed only that the pressure was radially uniform at the inlet. Migdal *et al.* and Brown and Ozcan mapped the inlet to minus infinity after Moretti, thus allowing the static conditions to be set equal to the stagnation conditions. Laval used extrapolation of the interior mesh points to determine the inlet mesh points, while Serra employed a characteristic scheme.

Prozan (in Ref. 1), Wehofer and Moger, Laval, and Brown and Ozcan used an extrapolation technique to compute the wall mesh points. Migdal *et al.* employed a characteristic scheme after Moretti to compute the wall mesh points, while Serra applied a reflection technique. For the converging nozzle problem to be properly posed, an exhaust jet calculation must be included. Wehofer and Moger used an extrapolation procedure to compute the exhaust jet boundary mesh points, while Brown and Ozcan employed a characteristic scheme after Moretti.

All of the above authors used extrapolation to compute the exit mesh points when the flow was supersonic, since any errors incurred would be swept out of the mesh. Serra employed a characteristic scheme when the exit flow was subsonic.

1.3 Choice of a Method

The lengthy computational times associated with time-dependent calculations are usually caused by inefficient numerical schemes or poor treatment of boundaries, resulting in the requirement for excessively fine computational meshes (see Refs. 8 and 9). A technique for a much more efficient calculation of the interior and boundary mesh points will be discussed here.

The computation of steady flows by a time-dependent method differs from ordinary initial-value problems in that the initial data and much of the transient solution have a negligible effect on the final or steady solution. Therefore, accuracy is important only for the asymptotic state, and special attention to intermediate efficiency will result in reasonable computational times. For this reason, interior mesh points can be computed by using a very efficient finite-difference scheme, as opposed to those less efficient finite-difference or characteristic schemes that achieve high accuracy at every step.

In the class of finite-difference schemes, the two-step methods such as the MacCormack and the two-step Lax-Wendroff schemes are more efficient than the original Lax-Wendroff scheme, especially if the governing equations are in conservation form. Moretti showed that using the equations of motion in conservation form decreased efficiency and ease of programming while only slightly increasing the accuracy of shock calculations. The use of an explicit artificial viscosity term for shock-free flows also decreases efficiency and was shown to be physically unjustified. In addition, such increases in the numerical dissipation can often destroy the weak shock structure of transonic flows. Therefore, the MacCormack scheme with the equations of motion in nonconservation form is used to calculate the interior mesh points. An explicit artificial viscosity term was included for

shock computations only. Remember that the implicit dissipation always present as an effect of truncation terms assures numerical stability for the shock-free flow results.

The boundary mesh points, while making up only a small part of the total mesh points, must be handled most accurately, because of the flowfield's sensitivity to precise boundary geometry. Moretti and Abbett showed that reflection, extrapolation, and one-sided difference techniques for computing solid wall boundaries give poor results and should be avoided. Therefore, the wall and centerbody mesh points are computed using a characteristic scheme. A characteristic scheme is also used to calculate the exhaust jet boundary mesh points.

In the case of the nozzle inlet mesh points for subsonic flow, the use of extrapolation techniques and the assumption of one-dimensional flow presume the form of the solution and in many cases are physically unjustified. On the other hand, a characteristic scheme could be used to calculate the inlet mesh points. While the stagnation pressure and temperature are assumed to remain constant at the inlet in a characteristic scheme (not necessarily the case for unsteady flow), this assumption would appear to be valid for the time-dependent calculation of steady flows. Moretti recommends mapping the inlet to minus infinity, thus allowing the static conditions to be set equal to the stagnation conditions. In theory, this appears to be the best approach, but it should be kept in mind that the infinite physical plane must be replaced by a finite computational plane. Also, this technique requires additional mesh points upstream of the nozzle inlet. It is not presently resolved as to whether the characteristic scheme approach used by Serra or the mapping-to-minus-infinity approach suggested by Moretti and employed by Migdal *et al.* and Brown and Ozcan is the best technique. To reduce the total number of mesh points to be computed, a characteristic scheme is used to compute the inlet mesh points. For supersonic flow, the inlet mesh points are set equal to specified values of velocity, pressure, and density, because in a supersonic stream the downstream conditions do not propagate upstream. Extrapolation is used to compute the exit mesh points when the flow is supersonic, since any errors incurred will be swept out of the mesh, and a characteristic scheme is employed when the flow is subsonic.

1.4 Equations of Motion

The appropriate non-conservation form of equations for two-dimensional, inviscid, isentropic, rotational flow are:

$$\rho_t + u\rho_x + v\rho_y + \rho u_x + \rho v_y + \epsilon\rho v/y = 0 \quad (1.1)$$

$$u_t + uu_x + vu_y + p_x/\rho = 0 \quad (1.2)$$

$$v_t + uv_x + vv_y + p_y/\rho = 0 \quad (1.3)$$

$$p_t + up_x + vp_y - a^2(\rho_t + u\rho_x + v\rho_y) = 0 \quad (1.4)$$

where ρ is the density, u is the axial velocity, v is the radial velocity, p is the pressure, a is the

local speed of sound, t is the time, x and y are the axial and radial coordinates, and the subscripts denote partial differentiation. The symbol ϵ is 0 for planar flow and 1 for axisymmetric flow.

1.4.1 Coordinate Transformation

The physical (x, y) plane is mapped into a rectangular computational plane (ζ, η) by the following coordinate transformation:

$$\zeta = x; \quad \eta = \frac{y - y_c(x)}{y_w(x, t) - y_c(x)}; \quad \tau = t \quad (1.5)$$

where $y_w(x, t)$ denotes the nozzle wall and exhaust jet boundary radius as a function of x and t and $y_c(x)$ denotes the nozzle centerbody radius as a function of x . These mapping functions must be single-valued functions of the x coordinate. In the (ζ, η, τ) coordinate system Equations (1.1)–(1.4) become:

$$\rho_\tau + u\rho_\zeta + \bar{v}\rho_\eta + \rho u_\zeta + \rho\alpha u_\eta + \rho\beta v_\eta + \epsilon\rho v/(y_c + \eta/\beta) = 0 \quad (1.6)$$

$$u_\tau + uu_\zeta + \bar{v}u_\eta + p_\zeta/\rho + \alpha p_\eta/\rho = 0 \quad (1.7)$$

$$v_\tau + uv_\zeta + \bar{v}v_\eta + \beta p_\eta/\rho = 0 \quad (1.8)$$

$$p_\tau + up_\zeta + \bar{v}p_\eta - a^2(\rho_\tau + u\rho_\zeta + \bar{v}\rho_\eta) = 0 \quad (1.9)$$

where

$$\beta = \frac{1}{y_w - y_c} \quad (1.10)$$

$$\alpha = -\beta \frac{\partial y_c}{\partial x} - \left(\frac{\partial y_w}{\partial x} - \frac{\partial y_c}{\partial x} \right) \quad (1.11)$$

$$\delta = -\eta\beta \frac{\partial y_w}{\partial t} \quad (1.12)$$

and

$$\bar{v} = \alpha u + \beta v + \delta \quad (1.13)$$

The fluid is assumed to be thermally and calorically perfect; that is, a constant ratio of specific heats is assumed.

1.4.2 Artificial Viscosity for Shock Computations

For shock computations, an artificial viscosity model of the form suggested by von Neumann-Richtmyer is used. This model, which has a term corresponding to all the viscous and thermal conduction terms in the Navier-Stokes equations, is shown below.

$$[\text{RHS Eq. (2)}] = (\lambda + 2\mu) \frac{\partial}{\partial x} \left(\frac{\partial u}{\partial x} \right) + \lambda \frac{\partial}{\partial x} \left(\frac{\partial v}{\partial y} \right) + \frac{\epsilon}{y} \left[(\lambda + \mu) \frac{\partial v}{\partial x} + \mu \frac{\partial u}{\partial y} \right] \quad (1.14)$$

$$[\text{RHS Eq. (3)}] = (\lambda + 2\mu) \frac{\partial}{\partial y} \left(\frac{\partial u}{\partial x} \right) + \lambda \frac{\partial}{\partial y} \left(\frac{\partial v}{\partial y} \right) + \mu \frac{\partial}{\partial x} \left(\frac{\partial u}{\partial y} + \frac{\partial v}{\partial x} \right) + \frac{\epsilon(\lambda + 2\mu)}{y} \left(\frac{\partial v}{\partial y} - \frac{v}{y} \right) \quad (1.15)$$

where $c_\mu = c_\lambda$ are nondimensional quantities that specify the distribution and amount of smoothing. **[Note: Complete artificial viscosity formulation to be filled from Section II of original report.]**

1.5 Numerical Method

The computational plane is divided into five sets of mesh points: interior, inlet, exit, wall and centerbody, and exhaust jet boundary.

1.5.1 Interior Mesh Points

The interior mesh points are computed using the MacCormack scheme, a second-order, non-centered, two-step, finite-difference scheme. Backward differences are used on the first step; forward differences are used on the second. The governing equations are left in non-conservation form. An explicit artificial viscosity term is used for shock computations. Centerline mesh points are computed by enforcing symmetry of the flow. For example, the finite-difference equations for Equation (1.1) for planar flow ($\epsilon = 0$) and no artificial viscosity are:

$$\bar{\rho}_{L,M}^{N+1} = \rho_{L,M}^N - \left[u_{L,M}^N \left(\frac{\rho_{L,M}^N - \rho_{L-1,M}^N}{\Delta x} \right) + v_{L,M}^N \left(\frac{\rho_{L,M}^N - \rho_{L,M-1}^N}{\Delta y} \right) + \rho_{L,M}^N \left(\frac{u_{L,M}^N - u_{L-1,M}^N}{\Delta x} \right) + \rho_{L,M}^N \left(\frac{v_{L,M}^N - v_{L,M-1}^N}{\Delta y} \right) \right] \quad (1.16)$$

$$\rho_{L,M}^{N+1} = 0.5 \left[\rho_{L,M}^N + \bar{\rho}_{L,M}^{N+1} - \left[\bar{u}_{L,M}^{N+1} \left(\frac{\bar{\rho}_{L+1,M}^{N+1} - \bar{\rho}_{L,M}^{N+1}}{\Delta x} \right) + \bar{v}_{L,M}^{N+1} \left(\frac{\bar{\rho}_{L,M+1}^{N+1} - \bar{\rho}_{L,M}^{N+1}}{\Delta y} \right) + \bar{\rho}_{L,M}^{N+1} \left(\frac{\bar{u}_{L+1,M}^{N+1} - \bar{u}_{L,M}^{N+1}}{\Delta x} \right) + \bar{\rho}_{L,M}^{N+1} \left(\frac{\bar{v}_{L,M+1}^{N+1} - \bar{v}_{L,M}^{N+1}}{\Delta y} \right) \right] \right] \quad (1.17)$$

where L and M denote axial and radial mesh points, respectively, N denotes the time step, and the bar denotes values calculated on the first step.

1.5.2 Inlet Mesh Points

The inlet mesh points for subsonic flow are computed using a second-order, reference-plane characteristic scheme. In this scheme, the partial derivatives with respect to η are computed in the initial-value and solution surfaces using non-centered differencing as in the MacCormack scheme. These approximations to the derivatives with respect to η are then treated as forcing terms and the

resulting system of equations is solved in the $\eta = \text{constant}$ reference planes using a two-independent-variable, characteristic scheme.

The boundary condition is the specification of the stagnation temperature and stagnation pressure. The use of a reference-plane characteristic scheme requires the specification of inlet flow angle as an additional boundary condition. The inlet flow angle can be approximately determined from the nozzle geometry. The equations relating the total and static conditions are:

$$p_T/p = [1 + (\gamma - 1)M^2/2]^{\gamma/(\gamma-1)} \quad (1.18)$$

$$T_T/T = 1 + (\gamma - 1)M^2/2 \quad (1.19)$$

where γ is the ratio of specific heats, M is the Mach number, T is the temperature, and the subscript T denotes the total conditions.

For supersonic flow, the inlet mesh points are set equal to specified values of velocity, pressure, and density.

1.5.3 Exit Mesh Points

For subsonic flow, a reference-plane characteristic scheme similar to the inlet scheme is used. The exit pressure is specified.

For supersonic flow, the exit mesh points are computed using linear extrapolation.

1.5.4 Wall and Centerbody Mesh Points

The wall and centerbody mesh points are computed using a reference-plane characteristic scheme. The wall and centerbody contours and therefore their slopes are specified. The boundary condition is given by:

$$v = u \tan(\theta) + \frac{\partial y_w}{\partial t} \quad (1.20)$$

where θ is the local wall or centerbody angle.

1.5.5 Exhaust Jet Boundary Mesh Points

The exhaust jet boundary mesh points are computed by the wall routine such that the pressure boundary condition:

$$p = p_{\text{ambient}} \quad (1.21)$$

is satisfied. This is accomplished by first assuming the shape of the jet boundary and then using the wall routine to calculate the pressure. Next, the jet boundary location is slightly changed and a second pressure is computed. By use of an interpolation procedure, a new jet boundary location

is determined. This interpolation-extrapolation procedure is then repeated at each point until the jet boundary pressure and the ambient pressure agree within some specified tolerance.

1.5.6 Time Step Control

The step size Δt is controlled by the well-known Courant or CFL condition, which can be expressed as:

$$\Delta t \leq \frac{1}{\left[(V + a) \left(\frac{1}{\Delta x^2} + \frac{1}{\Delta y^2} \right)^{1/2} \right]} \quad (1.22)$$

where V is the velocity magnitude. Using the coordinate transformation, Equation (1.22) becomes:

$$\Delta \tau \leq \frac{A}{\left[(V + a) \left(\frac{1}{\Delta \zeta^2} + \frac{\beta^2}{\Delta \eta^2} \right)^{1/2} \right]} \quad (1.23)$$

where the coefficient A was determined from actual calculations and varied between 0.4 and 1.6 depending on the geometry of the flow in question.

1.6 Overall Program Capabilities

The nozzle inlet flow, as well as the flow leaving the nozzle, may be either subsonic or supersonic. The flow may contain variations in stagnation temperature and stagnation pressure from streamline to streamline. The nozzle wall and centerbody geometries may be either one of two analytical contours or a completely general tabular contour. The program is capable of calculating the exhaust jet boundary for subsonic or supersonic flow. The initial data may be read in or calculated internally by the program. The internally computed data are calculated assuming one-dimensional, steady, isentropic flow with area change. The program output includes the coordinates, velocities, pressure, density, Mach number, temperature, mass flow, and axial thrust in both English and metric units.

1.7 Results and Discussion

The results presented here have been adopted from experimental validation work. The CDC 6600 computational times represent the central processor time not including compilation. So that these results can be compared with those of other investigators, the following table of relative machine speeds is provided:

Table 1.1: Relative Machine Speeds Compared to CDC 6600

Computer	Relative Machine Speed
IBM 7094	0.1
IBM 360/50	0.1
IBM 360/64	0.3
IBM 360/75	0.5
Univac 1108	0.5
CDC 6600	1.0

The validation cases are presented below.

1.7.1 Case 1: 45°-15° Conical Converging-Diverging Nozzle

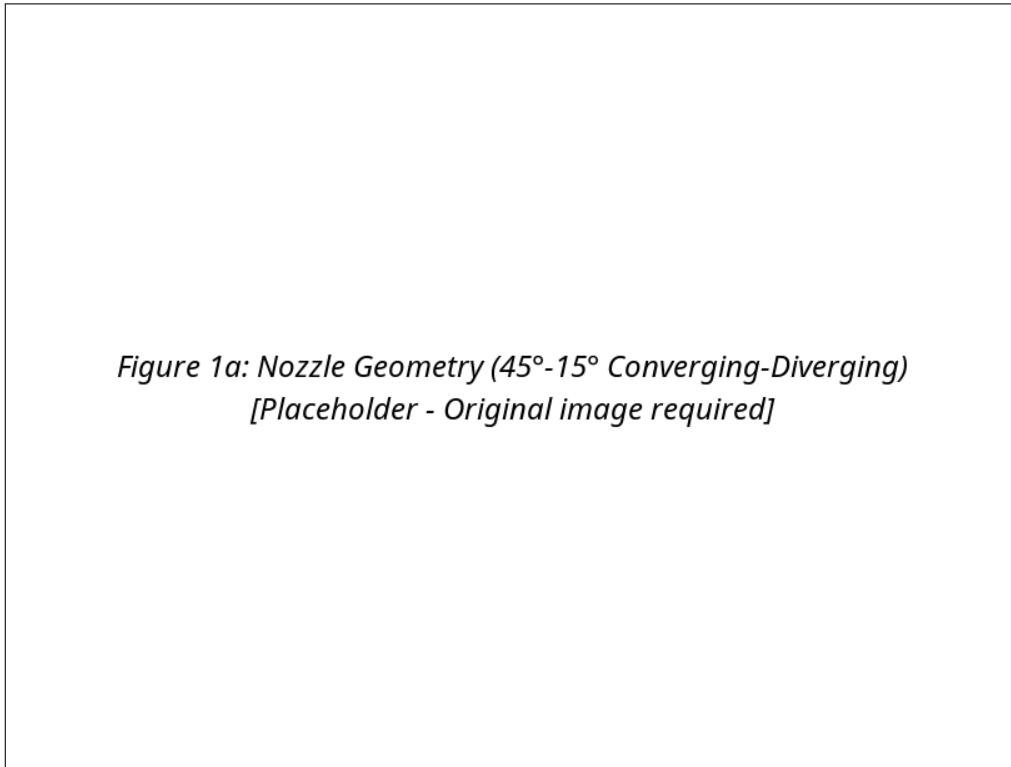
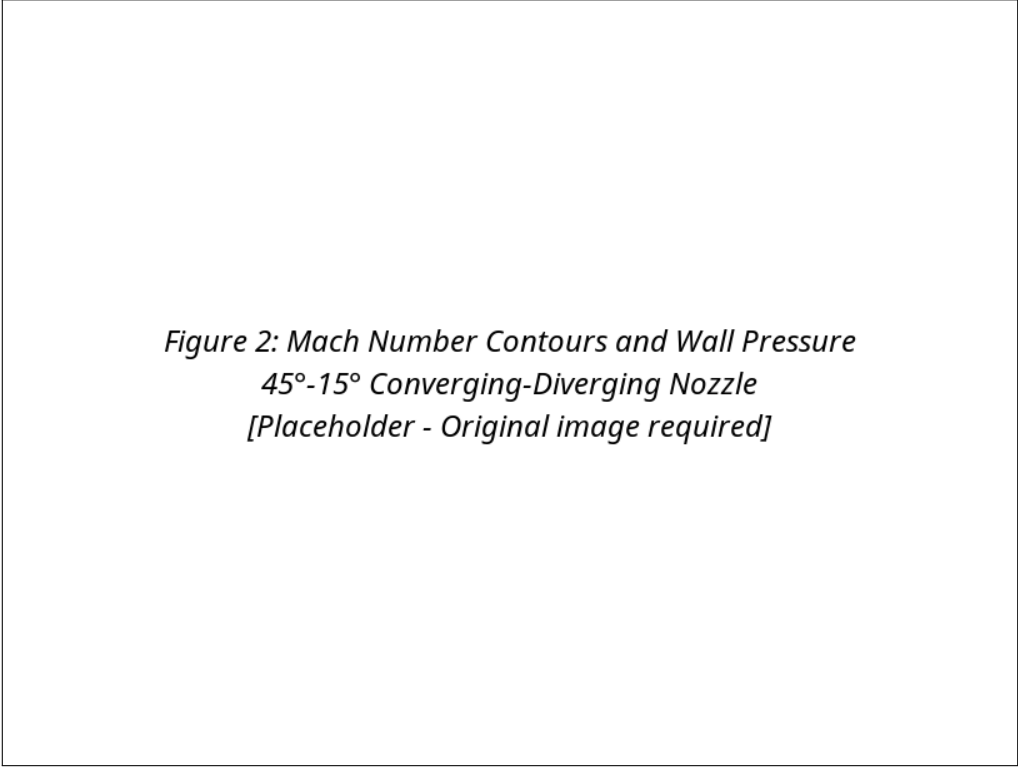


Figure 1.1: 45°-15° Converging-Diverging Nozzle Geometry

The present method was used to compute the steady-state solution for flow in the 45°-15° conical, converging-diverging nozzle. A 21×8 computational mesh required 301 time planes and a computational time of 35 seconds. The experimental data are those of Cuffel et al. (Ref. 2). The computed discharge coefficient is 0.983, compared with the experimental value of 0.985.



*Figure 2: Mach Number Contours and Wall Pressure
45°-15° Converging-Diverging Nozzle
[Placeholder - Original image required]*

Figure 1.2: Mach Number Contours and Wall Pressure Ratio for 45°-15° Conical Converging-Diverging Nozzle

There is good agreement with the experimental data. This case was also solved by other researchers including Prozan, Migdal, Laval, and Serra, with computational times ranging from 45 minutes to 2 hours on various computer systems.

1.7.2 Case 2: 15° Conical Converging Nozzle

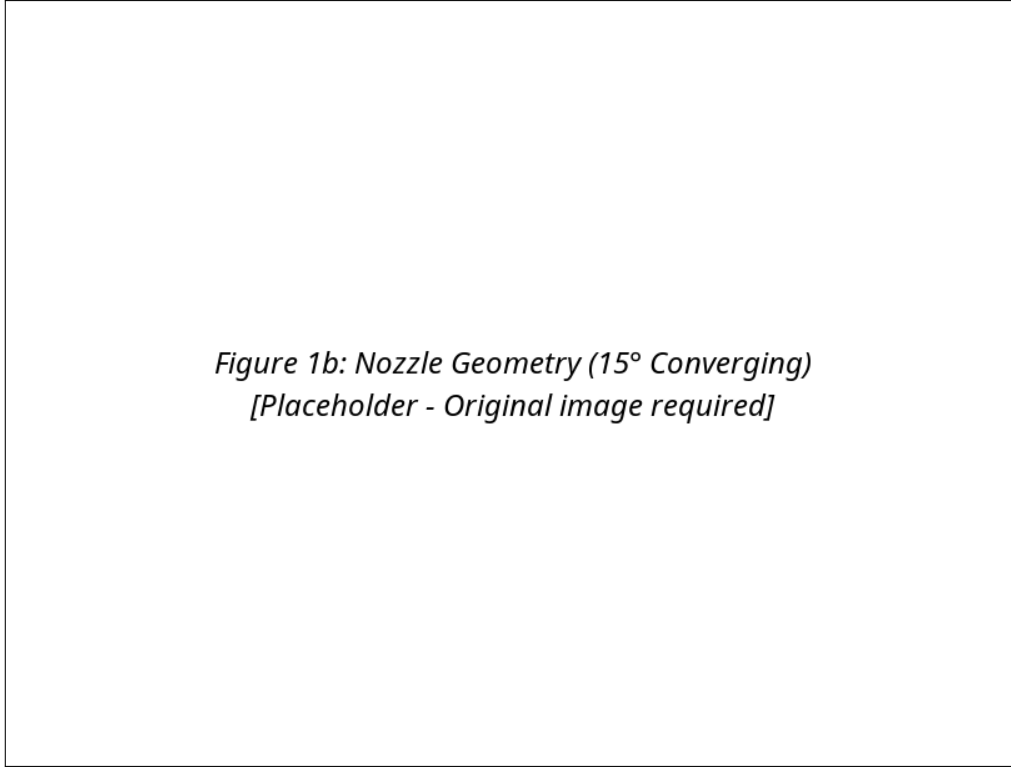
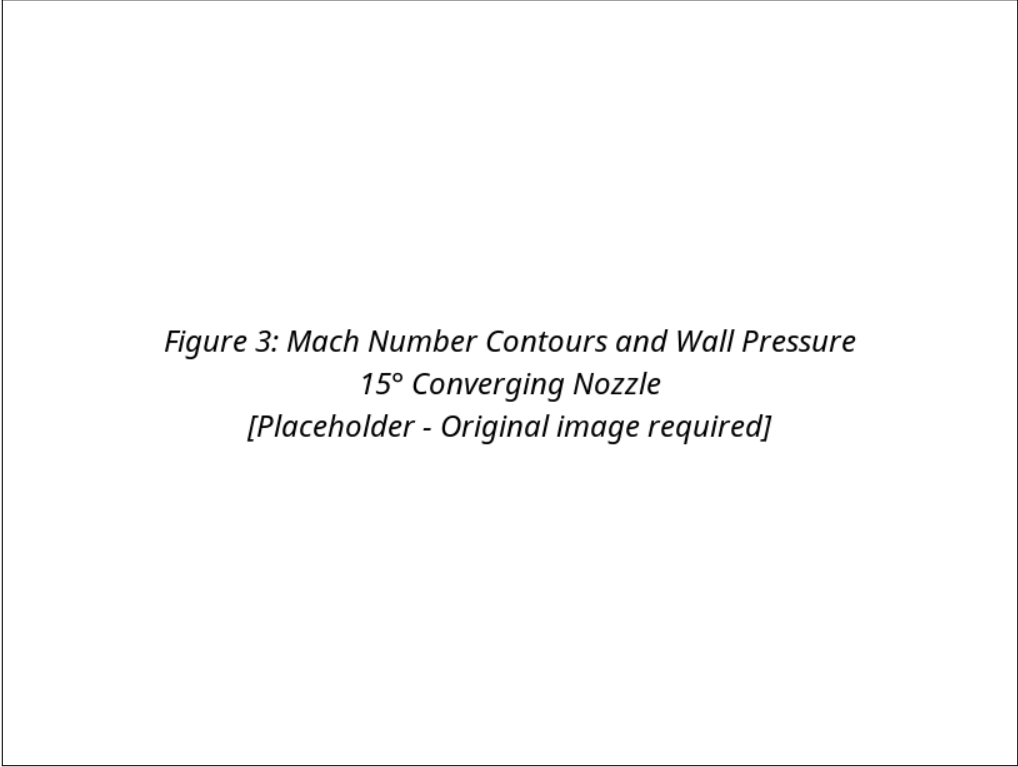


Figure 1.3: 15° Conical Converging Nozzle Geometry

The present method was also used to compute the steady-state flow in a 15° conical, converging nozzle. The nozzle geometry is shown in Figure 1.3. A 23×7 computational mesh required 249 time planes and a computational time of 29 seconds. The experimental data are those of Thornock (Ref. 17). The computed discharge coefficient is 0.957, compared with the experimental value of 0.960.



*Figure 3: Mach Number Contours and Wall Pressure
15° Converging Nozzle
[Placeholder - Original image required]*

Figure 1.4: Mach Number Contours and Wall Pressure Ratio for 15° Conical Converging Nozzle

There is good agreement with the experimental data. This case was also solved by Wehofer and Moger and Brown and Ozcan, with Wehofer and Moger requiring over 2 hours on an IBM 360/50 (47×11 mesh) and Brown and Ozcan requiring 17 minutes on an IBM 360/65 (20×6 mesh).

1.7.3 Case 3: 10° Conical Plug Nozzle

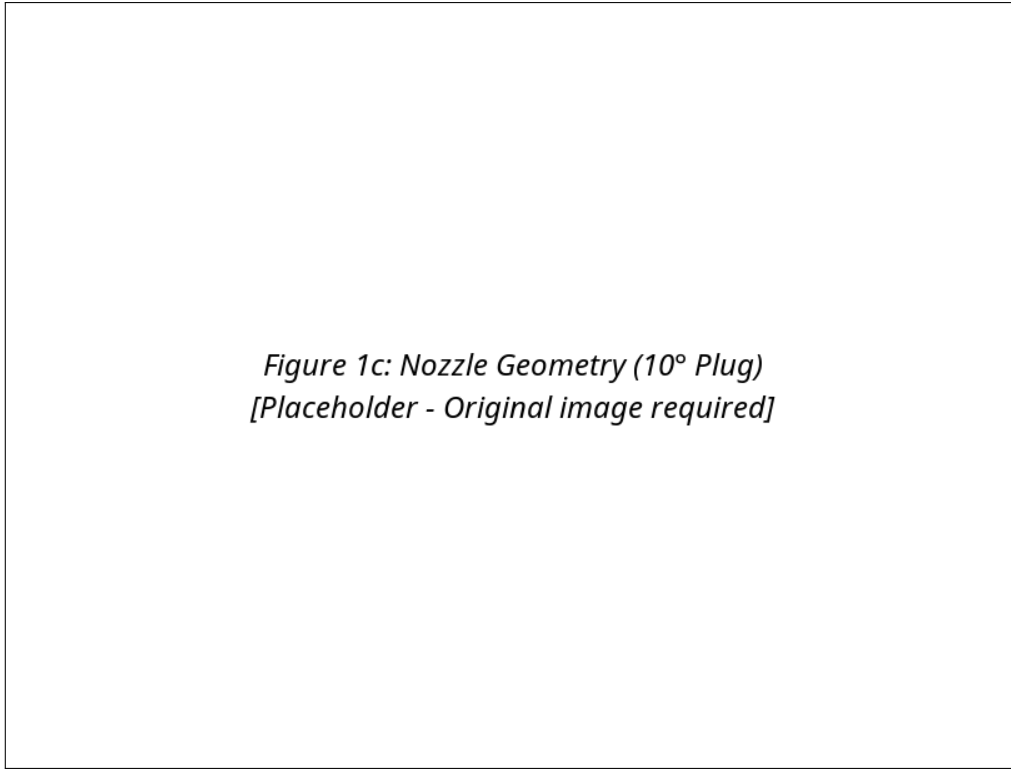


Figure 1.5: 10° Conical Plug Nozzle Geometry

Finally, the present method was used to calculate the flow in a 10° conical, plug nozzle. The nozzle geometry is shown in Figure 1.5. A 31×6 computational mesh required 327 time planes and a computational time of 52 seconds. The experimental data are those of Bresnahan and Johns (Ref. 18).

*Figure 4: Mach Number Contours and Plug Pressure
10° Conical Plug Nozzle
[Placeholder - Original image required]*

Figure 1.6: Mach Number Contours and Plug Pressure Ratio for 10° Conical Plug Nozzle

Again, there is good agreement with the experimental data. The author is unaware of any other time-dependent analyses of plug nozzles.

1.8 Concluding Remarks

A method of computing nozzle flows has been presented. A production-type computer program capable of solving a wide variety of nozzle flows has been developed. The program's accuracy was demonstrated by computing the steady flow in the three test cases above. The sub-one-minute computational times for these steady flows is considerably faster than for any of the earlier time-dependent techniques.

Chapter 2

Program Description and Usage

2.1 Program Structure

The NAP computer program consists of one main program, one function, and twelve subroutines. The program structure follows this execution flow:

1. **Program MAIN:** Initiates the run by reading input data, printing the program title and abstract, and converting units. Calls geometry subroutines and performs the main time-stepping loop.
2. **GEOM and GEOMCB:** Calculate the nozzle geometry for fixed wall and optional center-body configurations.
3. **ONEDIM:** Calculates the one-dimensional isentropic initial-value surface using a Newton-Raphson scheme to find Mach numbers from area ratios.
4. **MAP:** Calculates mapping functions that transform the physical plane to a rectangular computational plane.
5. **INTER:** Performs the interior mesh calculations using the MacCormack finite-difference scheme.
6. **INLET, WALL:** Implement boundary conditions using second-order characteristic-based schemes.
7. **MASFLO:** Calculates mass flow and thrust during the solution.
8. **PLOT:** Generates film plots of solution contours and velocity vectors.

2.2 Input Data Format

Input data are provided via Fortran NAMELIST format. The primary input namelists are:

NAMELIST /CNTRL/: Control Parameters

Key parameters controlling the simulation:

LMAX, MMAX Grid dimensions (ξ and η directions)

NMAX Maximum number of time steps

NPRINT Print frequency (0 = final solution only, $n > 0$ = every n steps)

TCONV Convergence criterion for steady-state detection

FDT Frequency for time-step recalculation

TSTOP Simulation stop time

GAMMA Specific heat ratio ($\gamma = 1.4$ for air)

RGAS Gas constant (53.35 for air in English units)

NASM Number of inlet stagnation point profiles (1 or > 1 for variable inlet conditions)

IUNIT Unit conversion flag (0 = English, 1 = SI)

NAMELIST /GEMTRY/: Geometry Parameters

Nozzle geometry definition:

NDIM Dimension flag (0 = 2D axisymmetric, 1 = 2D Cartesian)

NGEOM Geometry type (1 = converging, 2 = converging-diverging, 3 = plug)

XI, XE Inlet and exit axial coordinates

RI, RE Inlet and exit radii

RCI, RCT, RCE Centerbody inlet, throat, and exit radii (if centerbody present)

ANGI, ANGE Inlet and exit half-angles (degrees)

NWPTS Number of wall definition points

NAMELIST /BC/: Boundary Conditions

Inlet boundary condition parameters:

PT Stagnation pressure profile (array of NASM values)

TT Stagnation temperature profile (array of NASM values)

THETA Inlet flow angle profile (degrees)

PE Exit static pressure

NSTAG Stagnation profile flag (0 = uniform, > 0 = radial variation)

ISUPER Supersonic inlet flag (0 = subsonic, 1 = supersonic)

2.3 Output Description

The program produces output in three forms:

1. Printed Output

ASCII output containing:

- Program header and version information
- Echo of input parameters (CNTRL, GEMTRY, BC namelists)
- Initial geometry and one-dimensional surface calculations
- Iteration history with time, time-step size, and convergence measures (if requested)
- Final solution statistics including mass flow, momentum, and thrust

2. Film Plots

Vector plots and contour plots on graphics film (if $NPLOT \geq 0$):

- Velocity vectors at each solution time
- Contours of Mach number, pressure, density
- Wall streamline positions

3. Punched Card Output (Optional)

Fortran unformatted binary restart deck for continuing previous runs.

2.4 Sample Calculations

Three nozzle geometries have been analyzed and serve as test cases:

Case 1: Converging-Diverging Nozzle

A 45°–15° converging-diverging geometry with uniform inlet conditions at stagnation pressure $P_T = 13.78$ psia and temperature $T_T = 530$ °R. Results show excellent agreement with one-dimensional theory at the throat and quasi-2D behavior in the diverging section.

Case 2: Converging Nozzle

A simple 15° converging geometry with identical inlet conditions. Used to test subsonic inlet conditions and convergent-only nozzles.

Case 3: Plug Nozzle

A complex plug nozzle configuration with variable centerbody. Comparison with experimental data of Bresnahan and Johns (Reference 18) shows good agreement in gross features including shock structure at off-design conditions.

For detailed sample input and output listings, refer to the original NAP documentation. The program is controlled entirely via namelist input which provides flexibility for analyzing various nozzle configurations and inlet conditions.

Case No. 1 Input Deck (Converted from Original Fig. 7)

Editorial conversion note: Original Figure 7 is a compact data-deck listing. In this conversion, it is rendered as a cleaned text listing for readability and reuse.

```

1 CASE NO. 1 - CONVERGING-DIVERGING NOZZLE (45 DEG INLET, 15 DEG EXIT)
2
3 $CNTRL LMAX=21,MMA=8,NMAX=400,TCONV=0.005,FDT=1.6 $
4 $IVS $
5 $GEMTRY
   NGEOM=2,XI=0.31,RI=2.5,RT=0.8,XE=4.05,RCI=0.8,RCT=0.5,ANGI=44.88,ANGE=15.0 $
6 $GCBL $
7 $BC PT=70.0,TT=80.0 $

```

Case No. 1 Output (Converted from Original Fig. 8)

Editorial conversion note: In the original report, Figure 8 is a multi-page line-printer output block (run progression plus solution-surface table). In this conversion, it is represented as tabulated/listing content for readability and traceability.

Run Progression (from Fig. 8 pages 22–23)

1	10
2	20
3	30
4	40
5	50
6	60
7	70
8	80
9	90
10	100
11	110
12	120
13	130
14	140
15	150
16	160
17	170
18	180
19	190
20	200
21	210
22	220
23	230
24	240
25	250
26	260
27	270
28	280
29	290
30	300

Solution Surface Output Listing (Original Fig. 8, pages 24–27)

Note: A structured OCR-normalized table is provided below with full 21×8 row coverage (168 rows). Because this scan region is degraded, a subset of entries required OCR interpretation of ambiguous characters; raw extraction files are retained under `docs/conversion/fig8/` for auditability.

Table 2.2: Case No. 1 solution-surface output (Fig. 8) converted to structured table from OCR.

L	M	X (in)	Y (in)	U (fps)	V (fps)	P (psia)	RHO	Q (fps)	Mach	T (F)
1	1	0.3100	0.0300	144.3320	-0.0000	69.2160	0.347087	144.3320	0.1269	78.2651
1	2	0.3100	0.3571	141.7219	-0.0000	69.2440	0.347187	141.7219	0.1246	78.3272
1	3	0.3100	0.7143	128.2803	-0.0000	69.3801	0.347675	128.2883	0.1128	78.6293
1	4	0.3100	1.0714	109.3612	-0.0000	69.5491	0.348279	109.3612	0.0961	79.0040
1	5	0.3100	1.4286	86.4350	-0.0000	69.7181	0.348884	86.4350	0.0759	79.3778
1	6	0.3100	1.7857	62.2730	-0.0000	69.8536	0.349368	62.2730	0.0547	79.6771
1	7	0.3100	2.1429	39.9999	-0.0000	69.9396	0.349675	39.9999	0.0351	79.8668
1	8	0.3100	2.5000	19.8533	-0.0000	69.9851	0.349838	19.8533	0.0174	79.9672
2	1	0.4970	0.000	146.0691	0.0000	69.2904	0.347288	146.0691	0.1284	78.5319
2	2	0.4970	0.3540	143.4627	-4.4461	69.3196	0.347388	143.5516	0.1262	78.6038
2	3	0.4970	0.7083	130.0268	-8.4555	69.4507	0.347845	130.3006	0.1145	78.9123
2	4	0.4970	1.0619	110.9310	-10.9369	69.6135	0.348395	111.4688	0.0979	79.3234
2	5	0.4970	1.4159	67.8029	-11.7973	69.7741	0.348920	88.5919	0.0778	79.7545
2	6	0.4970	1.7699	63.3199	-10.8059	69.8959	0.349328	64.2353	0.0564	80.0651
2	7	0.4970	2.1239	40.7250	-6.3105	69.9730	0.349708	41.5643	0.0365	60.0739
2	8	0.4970	2.4778	20.2377	-4.6653	70.0063	0.349850	20.8143	0.0183	80.1108
3	1	0.6840	0.0008	153.2220	0.0000	69.1441	0.346842	153.2228	0.1347	78.0858
3	2	0.6840	0.3439	150.4176	-8.0795	69.1721	0.346941	150.6344	0.1325	78.1501
3	3	0.6840	0.6878	136.5817	-15.2234	69.2963	0.347379	137.4275	0.1208	78.4365
3	4	0.6840	1.0317	116.4617	-19.6710	69.4572	0.347946	118.1113	0.1038	78.8078
3	5	0.6840	1.3755	92.1280	-21.2109	69.6187	0.348556	94.5382	0.0631	79.1154
3	6	0.6840	1.7194	66.2422	-19.7190	69.7494	0.349177	69.1149	0.0607	79.1672
3	7	0.6840	2.0633	42.2740	-15.8931	69.8388	0.349479	45.1628	0.0397	79.3908
3	8	0.6840	2.4072	21.3548	-11.2935	69.8902	0.349784	24.1572	0.0212	79.3180
4	1	0.8710	0.0800	165.0230	0.0000	69.0375	0.346366	165.0230	0.1451	77.9945
4	2	0.8710	0.3243	161.8471	-12.7268	69.0716	0.346482	162.3467	0.1428	78.0821
4	3	0.8710	0.6487	148.1388	-24.0734	69.2124	0.346970	150.0821	0.1319	78.4187
4	4	0.8710	0.9734	127.6638	-31.8727	69.4034	0.347638	131.5824	0.1156	78.8667
4	5	0.8710	1.2973	102.9220	-35.8430	69.6042	0.348388	108.9846	0.0957	79.2632
4	6	0.8710	1.6217	76.1942	-35.2401	69.7817	0.348968	83.9523	0.0737	79.7398
4	7	0.8710	1.9460	51.5053	-32.0512	69.9154	0.349415	60.6636	0.0532	80.0812
4	8	0.8710	2.2703	29.4211	-28.9397	70.0166	0.349004	41.2688	0.0362	81.4995
5	1	1.0580	0.0000	184.0563	0.0000	68.7930	0.345583	184.0563	0.1620	77.3041
5	2	1.0580	0.2977	180.4888	-16.5732	68.8270	0.345704	181.2481	0.1595	77.3811
5	3	1.0580	0.5955	167.0840	-31.6930	68.9608	0.346189	170.0632	0.1496	77.6716

L	M	X (in)	Y (in)	U (fps)	V (fps)	P (psia)	RHO	Q (fps)	Mach	T (F)
5	4	1.0580	0.8932	146.2731	-42.9213	69.1497	0.346899	152.4403	0.1341	78.0406
5	5	1.0580	1.1909	120.6574	-49.3831	69.3592	0.347728	130.3722	0.1146	78.3850
5	6	1.0580	1.4887	92.4029	-49.9027	69.5637	0.348495	105.0170	0.0923	78.7833
5	7	1.0580	1.7864	66.3097	-46.6519	69.7342	0.349073	81.0763	0.0712	79.2094
5	8	1.0580	2.0841	45.6459	-45.4552	69.8646	0.351093	64.4183	0.0567	77.2645
6	1	1.2450	0.0000	2.086485	0.0000	68.3644	0.343931	208.6465	0.1838	76.5209
6	2	1.2450	0.2711	204.5468	-20.3330	68.4076	0.344081	205.5549	0.1810	76.6265
6	3	1.2450	0.5423	191.1932	-39.2038	68.5551	0.344612	195.1712	0.1718	76.9548
6	4	1.2450	0.8134	169.6380	-54.0883	68.7762	0.345419	178.0522	0.1567	77.4270
6	5	1.2450	0.8717	217.0973	-98.7450	67.6666	0.341360	238.4991	0.2103	75.0325
6	6	1.2450	1.3557	111.8738	-65.9052	69.3100	0.347214	129.8432	0.1141	78.7985
6	7	1.2450	1.6268	83.9892	-64.4311	69.5343	0.348057	105.8563	0.0930	79.2339
6	8	1.2450	1.8979	60.8833	-60.6288	69.8047	0.348123	85.9223	0.0753	81.2287
7	1	1.4320	0.0000	244.4324	0.0000	67.8755	0.342319	244.4324	0.2155	75.1934
7	2	1.4320	0.2445	239.7789	-25.3265	67.9268	0.342507	241.1127	0.2126	75.3031
7	3	1.4320	0.4091	226.5992	-49.2547	68.0787	0.343078	231.8906	0.2044	75.6076
7	4	1.4320	0.733	204.3100	-69.2211	68.3138	0.343971	215.7177	0.1901	76.0607
7	5	1.4320	0.9781	174.960	-82.8765	68.6031	0.345044	193.5836	0.1705	76.6578
7	6	1.4320	1.2226	140.4528	-87.7474	68.9166	0.346251	165.6098	0.1458	77.2313
7	7	1.4320	1.9672	107.5888	-85.8497	69.2024	0.347372	137.6427	0.1211	77.7176
7	8	1.4320	1.7117	79.2957	-78.9145	69.5794	0.349485	111.8364	0.0904	77.3792
8	1	1.6190	0.0000	289.8529	0.0000	66.7858	0.338237	269.8529	0.2561	72.9549
8	2	1.6190	0.2179	284.0910	-20.5977	66.3547	0.338482	286.1237	0.2526	73.1195
8	3	1.6190	0.9359	271.8205	-56.1340	67.0230	0.339096	277.5561	0.2451	73.4932
8	4	1.6190	0.6538	248.8164	-80.3206	67.2999	0.340096	261.4613	0.2300	74.1227
8	5	1.6190	0.8717	217.0973	-98.7450	67.6666	0.341360	238.4991	0.2103	75.0325
8	6	1.6190	1.911	1.0896	178.2156	-108.4309	60.1022	0.342072	208.6098	0.1838
8	7	1.6190	1.376	139.1469	-110.3320	68.5385	0.344469	177.5810	0.1563	77.0466
8	8	1.6190	1.5255	103.3920	-102.9598	69.3680	0.345942	145.9131	0.1279	81.2318
9	1	1.8060	0.0000	353.4364	0.0000	65.5800	0.334062	353.4364	0.3132	69.8416
9	2	1.8060	0.1913	398.2941	-34.5414	65.6636	0.334394	349.9529	0.3101	70.0227
9	3	1.8060	0.3826	336.6731	-68.5560	65.8359	0.335051	343.5825	0.3043	70.3717
9	4	1.8060	0.5790	314.9155	-100.6910	66.1349	0.336165	330.6213	0.2927	70.9827
9	5	1.8060	0.7653	282.8929	-128.3042	66.5481	0.337729	310.6290	0.2748	71.8578
9	6	1.8060	0.9566	243.3598	-147.2737	67.0462	0.339698	201.8907	0.2491	72.7483
9	7	1.8060	1.1979	192.3554	-154.1167	67.6139	0.341612	246.2463	0.2174	73.9216
9	8	1.8060	1.3393	133.7156	-133.1567	68.7510	0.346700	188.7076	0.	1664
10	1	1.9930	0.0000	436.8635	0.0000	62.9419	0.324223	436.8635	0.3893	63.9908

L	M	X (in)	Y (in)	U (fps)	V (fps)	P (psia)	RHO	Q (fps)	Mach	T (F)
10	2	1.9930	0.1697	432.1713	-35.7768	63.0217	0.324508	433.6496	0.3864	64.1950
10	3	1.9930	0.3299	423.3175	-71.8588	63.1559	0.324995	429.0775	0.3822	64.5243
10	4	1.9930	0.4992	404.2725	-107.9751	63.4042	0.325878	410.4434	0.3725	65.1589
10	5	1.9930	0.6589	373.7604	-142.2246	63.7957	0.327237	399.9133	0.3556	66.207-
10	6	1.9930	0.8236	328.4187	-171.0434	64.2934	0.329012	370.6602	0.3292	67.4514
10	7	1.9930	0.9803	264.4466	-165.0507	65.0844	0.331875	322.7626	0.2862	69.3341
10	8	1.9930	1.1530	173.3843	-172.3608	65.8445	0.333145	244.2672	0.2157	73.4752
11	1	2.1800	0.0000	549.3454	0.0000	59.7185	0.312647	549.3454	0.4935	55.5636
11	2	2.1800	0.1381	546.9007	-36.8068	59.7792	0.312892	548.2758	0.4925	55.6038
11	3	2.1800	0.2762	543.1126	-79.2687	59.8393	0.313169	548.8669	0.4930	55.7461
11	4	2.1800	0.4149	534.6434	-124.1219	59.9824	0.313805	546.8623	0.4929	55.9318
11	5	2.1800	0.5525	510.7335	-174.8851	60.2570	0.314978	547.4206	0.4914	56.3641
11	6	2.1800	0.6906	491.7599	-234.3386	60.7072	0.316985	544.7407	0.4888	56.9273
11	7	2.1800	0.8287	448.5535	-301.6778	61.5756	0.320739	540.5643	0.4844	58.1052
11	8	2.1800	0.9668	392.7955	-391.1038	61.5951	0.321452	554.2664	0.4972	57.1991
12	1	2.3670	0.0000	703.1835	0.0000	53.3954	0.286509	703.1835	0.6419	39.4042
12	2	2.3670	0.1195	705.4581	-30.2450	53.2708	0.280114	706.1062	0.6448	39.0606
12	3	2.3670	0.2389	711.906	-62.2785	52.9060	0.286719	714.6793	0.6533	36.0537
12	4	2.3670	0.3589	724.0761	-98.6945	52.2435	0.284196	730.7714	0.6692	36.1859
12	5	2.3670	0.9779	742.4833	-142.4451	51.1951	0.280222	756.0238	0.6945	33.1228
12	6	2.3670	0.5979	769.8530	-199.6792	49.5354	0.274167	795.3273	0.7347	27.6732
12	7	2.3670	0.7168	806.1904	-27.0574	47.0966	0.266002	851.4989	0.7946	17.8985
12	8	2.3670	0.8363	926.4913	-373.6663	40.8628	0.242460	999.0058	0.9555	-5.1008
13	1	2.5540	0.0000	867.9594	0.000	45.7225	0.250957	867.9594	0.8111	16.5731
13	2	2.5540	0.1193	873.9191	-18.6820	45.4031	0.257720	874.1188	0.8177	15.5161
13	3	2.5540	0.2266	887.6345	-37.8442	44.6459	0.254760	888.4409	0.8333	13.0183
13	4	2.5540	0.3929	914.6763	-56.5003	43.2240	0.249189	916.4197	0.8640	8.1917
13	5	2.5540	0.4571	956.2606	-72.8883	40.8916	0.239920	961.0287	0.9140	0.0397
13	6	2.5540	0.5719	1029.4131	-84.1867	37.0759	0.224347	1032.8498	0.9976	-13.9336
13	7	2.5540	0.6857	1148.3074	-75.4358	30.8926	0.197504	1150.7826	1.1425	-37.8082
13	8	2.5540	0.8000	1368.1926	-0.0522	22.7786	0.157750	1368.1926	1.4137	-70.2496
14	1	2.7410	0.0000	1057.9041	0.0000	36.0264	0.218543	1057.9041	1.0231	-15.0469
14	2	2.7410	0.1189	1067.8182	4.0614	35.4940	0.216255	1067.8292	1.0349	-16.9857
14	3	2.7410	0.2378	1089.2563	11.4080	34.3251	0.211179	1089.3160	1.0609	-21.2789
14	4	2.7410	0.3566	1130.3124	26.9803	32.2010	0.201817	1130.6344	1.1114	-29.3362
14	5	2.7410	0.4757	1193.1012	59.7158	28.9781	0.	187128	1194.5967	1.1919
14	6	2.7410	0.5946	1286.8774	125.7866	24.3779	0.	165031	1293.0103	1.3210
14	7	2.7410	0.7135	1418.7214	259.2371	18.2527	0.133085	1442.2116	1.5291	-69.8075

L	M	X (in)	Y (in)	U (fps)	V (fps)	P (psia)	RHO	Q (fps)	Mach	T (F)
14	8	2.7410	0.8325	1505.1920	403.3150	13.9588	0.108703	1558.2894	1.7074	-113.3946
15	1	2.9280	0.0000	1229.8892	0.0000	27.4288	0.179803	1229.8892	1.2364	-46.2460
15	2	2.9280	0.1261	1240.8800	32.2297	26.8607	0.177122	1241.2965	1.2516	-50.6718
15	3	2.9280	0.2522	1263.7631	67.2349	25.6518	0.171156	1265.4904	1.2843	-55.9399
15	4	2.9280	0.3782	1305.5670	116.1644	23.5457	0.161098	1310.7247	1.3462	-65.4968
15	5	2.9280	0.5043	1363.5225	186.1417	20.6584	0.146475	1376.1695	1.4388	-79.3179
15	6	2.9280	0.6304	1434.6074	283.8963	17.2610	0.128325	1462.4280	1.5657	-96.9372
15	7	2.9280	0.7565	1495.9417	379.2887	14.3497	0.111752	1543.2763	1.6910	-113.4100
15	8	2.9280	0.8826	1523.0676	408.1246	13.2341	0.104698	1576.7956	1.7414	-118.8190
16	1	3.1150	0.0000	1379.6124	2.0000	20.5366	0.146067	1379.6124	1.4447	-80.5066
16	2	3.1150	0.1332	1389.4688	57.2871	20.8520	0.143567	1390.6493	1.4611	-83.0061
16	3	3.1150	0.2665	1409.1486	116.9047	19.0318	0.118250	1413.9698	1.4964	-88.4294
16	4	3.1150	0.3997	1443.4828	190.1083	17.1330	0.129183	1455.9477	1.5607	-97.8448
16	5	3.1150	0.5330	1485.3850	276.8617	15.2534	0.117647	1510.9669	1.6476	-116.0416
16	6	3.1150	0.6662	1525.9339	367.4337	11.2773	0.106148	1569.5463	1.7425	-122.3815
16	7	3.1150	0.7994	1549.1994	421.6142	12.1092	0.098925	1605.5458	1.8019	-129.6003
16	8	3.1150	0.9327	1556.6257	417.0966	11.9876	0.097596	1611.5375	1.6055	-126.4656
17	1	3.3020	0.0000	1507.1046	0.0000	15.3430	0.118416	1507.1046	1.6440	-110.2754
17	2	3.3020	0.1404	1514.6529	77.2857	14.9837	0.116398	1516.6234	1.6597	-112.5417
17	3	3.3020	0.2808	1529.1232	155.3746	14.2210	0.112077	1536.9967	1.6942	-117.5164
17	4	3.3020	0.4212	1552.9446	242.7868	13.0110	0.105050	1571.8086	1.7537	-125.6938
17	5	3.3020	0.5616	1577.6060	331.4854	11.7060	0.097193	1612.0556	1.8239	-134.9123
17	6	3.3020	0.7020	1595.3473	403.6915	10.7335	0.091068	1645.6305	1.8821	-141.6704
17	7	3.3020	0.8424	1597.4021	431.4081	10.4723	0.089155	1654.6318	1.8956	-142.9533
17	8	3.3020	0.9828	1597.3052	427.9967	10.5899	0.089340	1653.6520	1.8859	-140.0550
18	1	3.4890	0.0000	1612.0423	0.0000	11.5688	0.096652	1612.0423	1.8295	-136.9232
18	2	3.4890	0.1476	1616.9281	91.6926	11.3290	0.095191	1619.5259	1.8433	-130.7641
18	3	3.4890	0.2951	1625.8324	182.2494	10.8049	0.091980	1636.0152	1.8742	-142.9296
18	4	3.4890	0.4427	1639.4093	276.0081	10.0197	0.087043	1662.4810	1.9240	-149.1947
18	5	3.4890	0.5902	1649.7370	358.9217	9.3066	0.082402	1688.3296	1.9726	-155.1532
18	6	3.4890	0.7378	1651.7627	412.3465	8.9842	0.080145	1702.4540	1.9965	-157.4273
18	7	3.4890	0.8853	1641.7622	427.9160	9.1585	0.061014	1696.6129	1.9813	-154.8662
18	8	3.4890	1.0329	1638.9946	439.1673	9.2411	0.081065	1696.8121	1.9733	-152.3077
19	1	3.6760	0.000	1704.0260	0.0000	8.8149	0.079430	1704.0260	2.0085	-160.4557
19	2	3.6760	0.1547	1706.2877	101.0366	8.6716	0.078460	1709.2764	2.0191	-161.7878
19	3	3.6760	0.3094	1709.9690	198.9763	8.3402	0.076304	1721.5068	2.0445	-164.9760
19	4	3.6760	0.4641	1714.2779	293.1834	7.8884	0.073245	1739.1680	2.0608	-169.3052
19	5	3.6760	0.6189	1712.3564	365.8767	7.6060	0.071250	1751.0084	2.1043	-171.8626

L	M	X (in)	Y (in)	U (fps)	V (fps)	P (psia)	RHO	Q (fps)	Mach	T (F)
19	6	3.6760	0.7736	1701.0635	403.9740	7.6934	0.071703	1749.1524	2.0967	-170.3935
19	7	3.6760	0.9283	1685.4753	420.4454	8.0075	0.071602	1737.1245	2.0679	-166.3468
19	8	3.6760	1.0830	1682.1090	450.7198	7.9878	0.073040	1741.4474	2.0677	-164.8157
20	1	3.8630	0.0000	1766.8661	0.0000	6.9239	0.066761	1766.8661	2.1542	-180.0643
20	2	3.8630	0.1619	1767.1843	105.9080	6.8519	0.066250	1770.3550	2.1615	-180.0404
20	3	3.8630	0.3237	1767.0745	206.8870	6.6633	0.064921	1779.1443	2.1605	-182.9639
20	4	3.8630	0.4856	1765.0631	297.6374	6.4524	0.063180	1769.9821	2.2026	-185.2113
20	5	3.8630	0.6475	1755.9466	360.1684	6.4495	0.063276	1792.5037	2.2045	-184.8822
20	6	3.8630	0.8094	1710.1859	391.2931	6.7291	0.065123	1783.6360	2.1787	-181.0971
20	7	3.8630	0.9712	1723.3741	415.9469	7.0217	0.066976	1772.8593	2.1499	-177.0232
20	8	3.8630	1.1331	1719.3424	460.6964	6.9351	0.065991	1779.9942	2.1559	-176.3400
21	1	4.0500	0.0000	1829.7062	-0.0000	5.0329	0.054092	1829.7062	2.3553	-206.8585
21	2	4.0500	0.1690	1828.0812	110.7794	5.0322	0.054012	1831.4345	2.3559	-206.5265
21	3	4.0500	0.3381	1824.1799	214.7978	4.9865	0.053538	1836.7827	2.3632	-208.6011
21	4	4.0500	0.5071	1815.8484	302.0914	5.0164	0.053515	1840.8054	2.3607	-206.9819
21	5	4.0500	0.6761	1799.5367	354.4601	5.2930	0.055301	1834.1141	2.3278	-201.6568
21	6	4.0500	1.1832	1756.5757	470.6731	5.8824	0.058942	1818.5410	2.2603	-190.6220
21	7	4.0500	1.0142	1761.2720	411.4465	6.0360	0.060351	1806.6933	2.2456	-190.0439
21	8	4.0500	1.1832	1756.5757	470.6731	5.8824	0.058942	1818.5410	2.2603	-190.6220

Case No. 3 Input Deck (Converted from Original Fig. 11)

Editorial conversion note: Original Figure 11 is a compact data-deck listing. In this conversion, it is rendered as a cleaned text listing so the namelist-style input can be copied and read directly.

```

1 CASE NO. 3 - CONVERGING-DIVERGING, PLUG NOZZLE (10 DEG CONE, PT/PE=3.29)
2
3 $CNTRL LMAX=31,MMAX=6,NMAX=400,TCONV=0.005,FDT=1.6 $
4 $TVS $
5 $GEMTRY NGEOM=1,XI=-4.440,XE=2.9600,RI=4.0,JFLAG=1,LJET=23 $
6 $GCBL NGCB=2,RICB=1.3,RTCB=3.365,RCICB=0.75,RCTCB=4.95,ANGICB=45.0,ANGECEB=10.0 $
7 $BC PT=100.0,TT=70.0,PE=30.4 $

```

Case No. 3 Output (Converted from Original Fig. 12)

Editorial conversion note: In the original report, Figure 12 is a multi-page line-printer output block (input summary, run progression, and solution-surface tables). In this conversion, it is represented as tabulated/listing content for readability and traceability rather than as a graphical figure.

Run Progression (from Fig. 12 continuation page)

Index	N
1	90
2	100
3	110
4	120
5	130
6	140
7	150
8	160
9	170
10	180
11	190
12	200
13	210
14	220
15	230
16	240
17	250
18	260
19	270
20	280
21	290
22	300
23	310
24	320
25	330
26	340

Input Summary Listing (Original Fig. 12, first page)

```

1 CO
2 IM          NAP,      A COMPUTER PROGRAM FOR THE COMPUTATION OF
          TNO"DIMENSIONAL, TIME-DEPENDENT,          INVISCID NOZZLE FLOW
3
4          BV MICHAEL C, CLINE,          t - I - LOS
          ALAMUS SCIENTIFIC LABORATORY
5
6
7
8          PROGRAM ABSTRACT -
9

```

THE EQUATIONS OF MOTION FOR TWO-DIMENSIONAL,
 TIME DEPENDENT, INVISCID FLOW IN A NOZZLE
 ARE SOLVED USING THE SECOND-ORDER, MACCORMACK,
 FINITE-DIFFERENCE SCHEME, THE FLUID IS ASSUMED
 TO BE A PERFECT GAS. ALL BOUNDARY CONDITIONS ARE
 COMPUTED USING A SECOND-ORDER, REFERENCE PLANE
 CHARACTERISTIC SCHEME, THE STEADY STATE SOLUTION IS
 OBTAINED AS THE ASYMPTOTIC SOLUTION FOR
 LARGE TIME, THE NOZZLES MAY BE EITHER CONVERGING,
 CONVERGING-DIVERGING, OR PLUG GEOMETRIES,

JOB TITLE -

CASE NO. 3 - CONVERGING-DIVERGING, PLUG NOZZLE (10
 DEG CONE, PT/PE<3,29)

CONTROL PARAMETERS -

L MAX*31	MMAX* 6	NMAX* 400	NPRINT* 0
TCONV* ,005	FDT * 1,60	NST AG*0	
NASH*1	IUNIT*0		
IUI = 1	IUO* 1	LEXsl	NCONVI* 1
TSTOP*1,00000	NID* 1	NPLOT* -1	
IPUNCH*0	ISUPER*0		
IAV*0	CAV* 5,0	XMU*1, 00	XLA*0,00
RKMU* ,50	CT A* ,50	LSS* 2	SMP*
,95	NST* 0		

FLUID MODEL -

THE RATIO OF SPECIFIC HEATS, GAMMA <1,4000 AND THE
 GAS CONSTANT, R 53,3500
 (FT-LBF/LBM-R)

FLOW GEOMETRY -

AXISYMMETRIC FLOW HAS BEEN SPECIFIED

NOZZLE GEOMETRY -

A CONSTANT AREA DUCT HAS BEEN SPECIFIED BY XI>
 -4,4400 (IN), R1< 4,0000 (IN),
 AND XE* 2,9600 (IN)

```

42      AN EXHAUST JET CALCULATION HAS BEEN REQUESTED,  THE
43      NOZZLE ENDS AT X*      ,7400 (IN),  THE
44      MESH POINTS L* 23 TO L* 31 ARE AN INITIAL
45      APPROXIMATION TO THE EXHAUST JET BOUNDARY.
46
47      A CIRCULAR-ARC, CONICAL CENTERBODY HAS BEEN SPECIFIED
48      BY XICB* -4,4400 (IN), RICB*  1,3000 (IN),
49      RTCB* 3.3650 (IN), XECB*    2,9600 (IN), RCICB*
50      ,7500 (IN), RCTCB* 4,9500 (IN), ANGICB* 45,00
51      (DEG) ,
52      AND ANGEGB* 10,00 (DEG). THE: COMPUTED VALUES ARE
53      XTCB* -.0143 (IN) AND RECB*  2,9170 (IN) ,
54
55      BOUNDARY CONDITIONS -
56
57      PT* 100.0000 (PSIA)      TT*      10,0000 (F)
58      THETA*      0.0000 (DEG)      PE*   30,4000
59      (PSIA)
60
61      N*   10
62      N*   20
63      N*   30
64      N*   40
65      Ns   50
66      N*   60
67      N*   70
68      N*   00
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Fig. 12 Case No.
3 output

Solution Surface Output Listing (Original Fig. 12, pages 38–41)

Note: OCR on pages 40–41 can read the surface number as “395” in the header, but the L column continues monotonically from 1 to 31 and the reported times are effectively unchanged; this block is treated as one continued solution-surface output. A structured OCR-normalized table is provided below with full 31×6 row coverage (186 rows). Some rows required OCR interpretation of ambiguous digits/symbols; raw extraction files are retained under `docs/conversion/fig12/` for auditability.

Table 2.4: Case No. 3 solution-surface output (Fig. 12) converted to structured table from OCR (high-confidence rows).

L	M	X (in)	Y (in)	U (fps)	V (fps)	P (psia)	RHO	Q (fps)	Mach	T (F)
1	1	-4.4404	1.3000	109.4812	-0.0000	99.3430	0.506883	109.4812	0.0971	69.0027
1	2	-4.4400	1.8400	106.0169	-0.0000	99.3831	0.507029	106.0169	0.0940	69.0638

L	M	X (in)	Y (in)	U (fps)	V (fps)	P (psia)	RHO	Q (fps)	Mach	T (F)
1	3	-4.4400	2.3830	212.9815	-0.0000	97.5259	0.500243	212.9815	0.1894	66.2199
1	4	-4.4400	2.9200	236.6861	-0.0000	96.9532	0.498143	236.6861	0.2107	65.3351
1	5	-4.4400	3.4600	276.8857	-0.0000	95.8470	0.494076	276.8857	0.2468	63.6156
1	6	-4.4400	4.0000	273.4428	-0.0000	95.9570	0.494481	273.4428	0.2437	63.7873
2	1	-4.1933	1.3417	107.4307	37.4142	99.2108	0.506393	113.7593	0.1009	68.8098
2	2	-4.1933	1.8734	104.0377	29.6151	99.2062	0.506362	108.1707	0.0960	68.8175
2	3	-4.1933	2.4350	214.2242	22.0820	97.4455	0.499962	215.3593	0.1915	66.0820
2	4	-4.1933	2.9367	237.1094	13.9520	96.9338	0.498071	237.5195	0.2114	65.3056
2	5	-4.1933	3.4683	277.6715	7.3298	95.8509	0.494092	277.7683	0.2476	63.6205
2	6	-4.1933	4.0000	274.0878	0.0000	95.9535	0.494468	274.0878	0.2443	63.7824
3	1	-3.9467	1.4851	112.9896	98.6735	98.7348	0.504664	150.0103	0.1332	68.0751
3	2	-3.9467	1.9881	119.2811	55.7369	99.1459	0.506168	131.6609	0.1168	68.6990
3	3	-3.9467	2.4911	230.3713	50.8948	97.1220	0.498801	235.9263	0.2099	65.5554
3	4	-3.9467	2.9940	245.1004	28.8405	96.7471	0.497390	246.7914	0.2197	65.0115
3	5	-3.9467	3.4970	284.0488	15.6501	95.6688	0.493423	284.4796	0.2537	63.3345
3	6	-3.9467	4.0000	279.0134	0.0000	95.8206	0.493979	279.0134	0.2487	63.5740
4	1	-3.7000	1.7293	133.0260	133.0260	97.9867	0.501924	188.1272	0.1672	66.9351
4	2	-3.7000	2.1835	152.2549	70.3057	98.2987	0.503055	167.7034	0.1490	67.4243
4	3	-3.7000	2.6376	251.8880	69.9268	96.4944	0.496505	261.4141	0.2328	64.5736
4	4	-3.7000	3.0917	258.5497	37.0252	96.2556	0.495578	261.1873	0.2327	64.2546
4	5	-3.7000	3.5459	293.1525	20.9197	95.3831	0.492369	293.8980	0.2622	62.8874
4	6	-3.7000	4.0000	287.4930	0.0000	95.5579	0.493011	287.4930	0.2564	63.1645
5	1	-3.4533	1.9750	165.0872	159.4938	97.2086	0.499081	229.5475	0.2042	65.7285
5	2	-3.4533	2.3800	189.7552	88.9515	97.4650	0.500033	209.5695	0.1864	66.1113
5	3	-3.4533	2.7850	274.0965	86.3530	95.8902	0.494289	287.3773	0.2562	63.6267
5	4	-3.4533	3.1900	275.5988	45.2297	95.7240	0.493625	279.2855	0.2490	63.4220
5	5	-3.4533	3.5950	305.6486	25.4583	95.0148	0.491013	306.7070	0.2738	62.3076
5	6	-3.4533	4.0000	300.2397	0.0000	95.1903	0.491657	300.2397	0.2679	62.5870
6	1	-3.2067	2.1978	205.2765	173.2552	96.0735	0.494909	268.6183	0.2394	63.9703
6	2	-3.2067	2.5582	229.6718	104.0684	96.2531	0.495595	252.1495	0.2247	64.2229
6	3	-3.2067	2.9187	297.7834	94.9364	95.0763	0.491294	312.5506	0.2790	62.3581
6	4	-3.2067	3.2791	296.9780	50.8083	94.9696	0.490842	301.2929	0.2689	62.2412
6	5	-3.2067	3.6396	321.7514	27.9020	94.4542	0.488941	322.9590	0.2885	61.4256
6	6	-3.2067	4.0000	317.4202	0.0000	94.6142	0.489529	317.4202	0.2835	61.6822
7	1	-2.9600	2.3929	246.1173	182.2747	94.9109	0.490626	306.2642	0.2734	62.1473
7	2	-2.9600	2.7143	267.6866	117.9731	95.0919	0.491347	292.5299	0.2611	62.3756
7	3	-2.9630	3.0357	323.2732	100.9577	94.2213	0.488127	338.6709	0.3027	61.0076
7	4	-2.9600	3.3571	321.5757	56.3837	94.1319	0.487751	326.4813	0.2918	60.9145

L	M	X (in)	Y (in)	U (fps)	V (fps)	P (psia)	RHO	Q (fps)	Mach	T (F)
7	5	-2.9600	3.6736	341.9079	29.8608	93.7507	0.486340	343.2094	0.3069	60.3106
7	6	-2.9600	4.0000	338.8474	0.0000	93.8859	0.486837	338.8474	0.3030	60.5294
8	1	-2.7133	2.5642	289.9157	188.6100	93.5079	0.485435	345.8682	0.3094	59.9300
8	2	-2.7133	2.8514	307.9821	127.9301	93.7199	0.486292	333.4952	0.2983	60.1912
8	3	-2.7133	3.1385	353.1446	104.0948	93.1129	0.484017	368.1668	0.3296	59.2511
8	4	-2.7133	3.4257	350.7214	60.2814	93.0513	0.483748	355.8642	0.3186	59.1957
8	5	-2.7133	3.7128	367.0750	30.9629	92.7906	0.482778	368.3786	0.3299	58.7817
8	6	-2.7133	4.0000	365.0832	0.0000	92.9035	0.483193	365.0832	0.3269	58.9667
9	1	-2.4667	2.7146	335.9522	191.6407	91.9380	0.479603	386.7687	0.3469	57.4175
9	2	-2.4667	2.9717	350.4988	134.3668	92.1960	0.480650	375.3716	0.3365	57.7394
9	3	-2.4667	3.2288	387.1967	105.6010	91.7925	0.479105	401.3389	0.3600	57.1358
9	4	-2.4667	3.4859	384.1126	63.0205	91.7586	0.478944	389.2481	0.3492	57.1185
9	5	-2.4667	3.7429	397.2721	31.6727	91.5882	0.478304	398.5327	0.3576	56.8485
9	6	-2.4667	4.0000	396.0271	0.0000	91.6830	0.478654	396.0271	0.3553	57.0061
10	1	-2.2200	2.8463	385.7795	192.0533	90.0749	0.472645	430.9412	0.3876	54.3954
10	2	-2.2200	3.0770	396.9902	137.6498	90.3795	0.473682	420.1767	0.3778	54.7871
10	3	-2.2200	3.3078	426.5248	105.6109	90.1440	0.472945	439.4053	0.3952	54.4635
10	4	-2.2200	3.5385	422.6905	64.3804	90.1463	0.472923	427.5654	0.3845	54.5002
10	5	-2.2200	3.7693	433.1660	31.9255	90.0516	0.472562	434.3409	0.3907	54.3518
10	6	-2.2200	4.0000	432.3408	0.0000	90.1354	0.472872	432.3408	0.3888	54.4940
11	1	-1.9733	2.9607	440.1175	189.7053	87.8573	0.464311	479.2614	0.4326	50.7368
11	2	-1.9733	3.1686	448.0872	136.0339	88.2164	0.465769	466.8662	0.4230	51.2182
11	3	-1.9733	3.3764	471.3597	104.1647	88.1287	0.465373	482.7320	0.4356	51.1456
11	4	-1.9733	3.5843	466.6714	64.4745	88.1745	0.465519	471.1042	0.4250	51.2510
11	5	-1.9733	3.7921	474.8778	31.7414	88.1431	0.465393	475.9374	0.4294	51.2060
11	6	-1.9733	4.0000	474.2174	0.0000	88.2221	0.465686	474.2174	0.4278	51.3423
12	1	-1.7267	3.0593	500.0783	184.4158	85.1723	0.454142	532.9986	0.4833	46.2146
12	2	-1.7267	3.2474	504.7398	135.5893	85.6008	0.455882	522.6344	0.4736	46.8202
12	3	-1.7267	3.4356	522.3541	101.1329	85.6519	0.456001	532.0542	0.4820	46.9892
12	4	-1.7267	3.6237	516.6715	63.2783	85.7507	0.456353	520.5320	0.4715	47.1831
12	5	-1.7267	3.8119	522.8704	31.0492	85.7759	0.456442	523.7915	0.4744	47.2336
12	6	-1.7267	4.0000	522.1891	0.0000	85.8554	0.456738	522.1891	0.4729	47.3748
13	1	-1.4800	3.1429	566.2529	175.5815	61.9333	0.441756	592.8501	0.5405	40.6179
13	2	-1.4800	3.3143	567.4408	130.0717	82.4540	0.443870	582.1578	0.5304	41.4001
13	3	-1.4800	3.4858	579.7939	96.2310	82.6443	0.444521	587.7255	0.5352	41.8215
13	4	-1.4800	3.6572	572.9430	60.6967	82.8076	0.445125	576.1491	0.5245	42.1299
13	5	-1.4800	3.8286	577.3073	29.7520	82.8869	0.445422	578.0735	0.5262	42.2757
13	6	-1.4800	4.0000	576.4726	0.0000	82.9705	0.445737	576.4726	0.5246	42.4267

L	M	X (in)	Y (in)	U (fps)	V (fps)	P (psia)	RHO	Q (fps)	Mach	T (F)
14	1	-1.2333	3.2125	639.0676	162.4290	78.0504	0.426727	659.3865	0.6054	33.6883
14	2	-1.2333	3.3700	636.5520	121.0465	78.6926	0.429342	647.9589	0.5943	34.7195
14	3	-1.2333	3.5275	643.9203	89.0877	79.0292	0.430567	650.0539	0.5958	35.4213
14	4	-1.2333	3.6850	635.7200	56.5499	79.2706	0.431485	638.2302	0.5847	35.8779
14	5	-1.2333	3.8425	638.3532	27.7308	79.4040	0.431995	638.9553	0.5852	36.1254
14	6	-1.2333	4.0000	637.2739	0.0000	79.4949	0.432340	637.2739	0.5835	36.2970
15	1	-0.9867	3.2685	718.4278	143.9795	73.4700	0.408725	732.7132	0.6786	25.1849
15	2	-0.9867	3.4148	711.9765	107.8905	74.2675	0.411994	720.1048	0.6660	26.5599
15	3	-0.9867	3.5611	714.5835	79.2171	74.7626	0.413865	718.9610	0.6642	27.5894
15	4	-0.9867	3.7074	704.8565	50.5751	75.0970	0.415164	706.6686	0.6524	28.2372
15	5	-0.9867	3.8537	705.8355	24.8312	75.2871	0.415905	706.2721	0.6518	28.6006
15	6	-0.9867	4.0000	704.4402	0.0000	75.3868	0.416293	704.4402	0.6500	28.7919
16	1	-0.7400	3.3115	803.7732	119.1704	68.1821	0.387547	812.5607	0.7607	14.8693
16	2	-0.7400	3.4492	793.2104	89.8965	69.1705	0.391649	798.2882	0.7456	16.7076
16	3	-0.7400	3.5869	791.2734	66.0987	69.8382	0.394253	794.0294	0.7408	18.1298
16	4	-0.7400	3.7246	779.9013	42.4700	70.2601	0.396011	781.0568	0.7280	19.0199
16	5	-0.7400	3.8623	779.2786	20.8999	70.5298	0.397005	779.5588	0.7262	19.5177
16	6	-0.7400	4.0000	777.5410	0.0000	70.6408	0.397439	777.5410	0.7242	19.7486
17	1	-0.4933	3.3417	893.9432	86.9769	62.2513	0.363220	898.1644	0.8519	2.6020
17	2	-0.4933	3.4734	879.1927	66.3566	63.4636	0.368344	881.6933	0.8340	5.0490
17	3	-0.4933	3.6050	873.0032	49.2059	64.3169	0.371780	874.3888	0.8254	6.9467
17	4	-0.4933	3.7367	859.9066	31.9340	64.8781	0.374068	860.4994	0.8113	8.1395
17	5	-0.4933	3.8683	857.7794	15.7672	65.1907	0.375342	857.9243	0.8083	8.7992
17	6	-0.4933	4.0000	855.6361	0.0000	65.3116	0.375837	855.6361	0.8059	9.0498
18	1	-0.2467	3.3595	987.1818	46.4557	55.8187	0.336066	988.2743	0.9522	-11.7118
18	2	-0.2467	3.4876	968.3735	36.6601	57.2806	0.342422	969.0721	0.9303	-8.4827
18	3	-0.2407	3.6157	958.2832	28.1325	58.3275	0.346782	958.6961	0.9179	-6.0113
18	4	-0.2467	3.7438	943.5326	18.7218	59.0161	0.349673	943.7183	0.9020	-4.4493
18	5	-0.2467	3.8719	939.9736	9.3489	59.3931	0.351253	940.0201	0.8976	-3.6020
18	6	-0.2467	4.0000	937.6150	0.0000	59.5321	0.351811	937.6150	0.8950	-3.2594
19	1	0.0300	3.3650	1081.9350	-3.0509	49.1261	0.306820	1081.0393	1.0612	-27.8336
19	2	0.0000	3.0920	1056.8199	0.3966	50.8097	0.310533	1058.8195	1.0300	-23.6353
19	3	0.0000	3.6190	1005.3991	2.5186	52.0880	0.319889	1005.3522	1.0172	-20.0885
19	4	0.0000	3.7960	1029.0988	2.6307	52.9016	0.323016	1029.0521	0.9990	-18.0950
19	5	0.0000	3.8730	1020.3133	1.0601	53.3012	0.325310	1020.3103	0.9932	-17.0236
19	6	0.0300	4.0000	1020.9992	0.0000	53.0719	0.325911	1020.9992	0.9897	-17.1525
20	1	0.2967	3.3581	1173.1200	-61.8580	42.3090	0.276131	1170.7501	1.1778	-46.0380
20	2	0.2967	3.0865	1106.0302	-42.2793	46.3023	0.285382	1108.8080	1.1003	-40.6080

L	M	X (in)	Y (in)	U (fps)	V (fps)	P (psia)	RHO	Q (fps)	Mach	T (F)
20	3	0.2967	3.6199	1132.0781	-26.9501	45.7752	0.291831	1132.3988	1.1227	-36.6236
20	4	0.2967	3.7033	1110.2692	-15.5120	46.7326	0.296129	1110.3772	1.1010	-30.0010
20	5	0.2967	3.8716	1107.7929	-6.9782	47.2613	0.298089	1107.8109	1.0932	-32.6288
20	6	0.2967	4.0000	1107.0192	0.0000	47.5001	0.299051	1107.0192	1.0913	-31.6133
21	1	0.9933	3.3389	1269.0032	130.7062	35.7235	0.200355	1275.7208	1.3101	-65.3967
21	2	0.9933	3.9711	1236.3910	-93.2116	37.9196	0.255030	1201.8900	1.2606	-58.6706
21	3	0.9933	3.6030	1219.8855	-62.9528	39.5789	0.202656	1221.5088	1.2360	-53.5867
21	4	0.9933	3.7356	1200.0219	-39.0606	40.7509	0.268366	1200.6576	1.2097	-50.0979
21	5	0.9933	3.8676	1193.6520	-20.1323	41.3780	0.271285	1193.8222	1.2002	-48.3097
21	6	0.9933	4.0000	1179.0275	0.0000	41.3050	0.271083	1179.0275	1.1867	-48.9338
22	1	0.7900	3.3072	1331.6916	205.2378	30.9096	0.220003	1307.0102	1.4120	-81.0062
22	2	0.7900	3.9058	1305.5170	155.2005	32.8860	0.230385	1310.7103	1.3663	-70.7132
22	3	0.7900	3.5893	1208.8579	111.5273	30.0362	0.238202	1293.6703	1.3360	-69.7912
22	4	0.7900	3.7229	1272.2609	-71.9720	35.0028	0.203108	1270.2950	1.3113	-66.9980
22	5	0.7900	3.6619	1267.2158	-32.7531	35.0110	0.205260	1267.6390	1.3026	-65.8919
22	6	0.7900	4.0000	1286.7778	0.0000	35.2207	0.201585	1286.7778	1.3233	-66.0902
23	1	0.9667	3.2609	1383.2720	203.9082	27.2665	0.201293	1000.6112	1.0980	-90.3500
23	2	0.9667	3.0123	1370.5310	187.5136	28.2071	0.206581	1387.2628	1.0731	-90.9272
23	3	0.9867	3.5596	1370.0887	128.3918	28.8860	0.209959	1380.0700	1.0609	-86.6062
23	4	0.9667	3.7070	1372.0803	-67.7226	29.0913	0.210905	1373.7506	1.0520	-87.6901
23	5	0.9867	3.8503	1375.5103	-12.7883	29.2328	0.211388	1375.5738	1.0520	-86.7353
23	6	0.9867	4.0016	1355.3965	9.0228	30.0018	0.217203	1355.0266	1.0227	-82.2707
24	1	1.2333	3.2219	1030.0370	252.8596	20.1025	0.180583	1056.1601	1.5809	-106.9657
24	2	1.2333	3.3752	1036.0195	196.1252	20.5272	0.186710	1009.7069	1.5706	-105.0328
24	3	1.2333	3.5291	1002.6305	135.8278	20.8330	0.188222	1009.0107	1.5660	-103.8817
24	4	1.2333	3.6829	1029.7078	-69.7732	25.6828	0.192670	1032.5235	1.5006	-100.2039
24	5	1.2333	3.8367	1397.0981	-69.0626	27.8136	0.203959	1399.2230	1.0876	-91.9200
24	6	1.2333	3.9905	1350.6985	-61.0675	30.3938	0.217202	1356.0702	1.0230	-82.2982
25	1	1.9600	3.1779	1001.9167	261.3019	21.3229	0.168627	1500.7777	1.6616	-116.6920
25	2	1.9600	3.3359	1082.1819	207.8316	21.7522	0.171088	1096.6810	1.6081	-116.8276
25	3	1.9600	3.0939	1070.3210	165.8729	22.8291	0.177072	1083.6231	1.6220	-112.0096
25	4	1.9600	3.6519	1033.9630	109.1605	25.1620	0.189852	1001.7000	1.5509	-102.2670
25	5	1.9800	3.8098	1392.5720	137.2705	27.8310	0.200008	1399.3220	1.0877	-91.8057
25	6	1.0800	3.9678	1309.1683	120.2698	30.9025	0.217206	1350.8790	1.0221	-82.2663
26	1	1.7267	3.1309	1509.8281	266.2235	20.0212	0.161080	1533.1197	1.7096	-125.3505
26	2	1.7267	3.2992	1099.0577	251.5096	20.6565	0.165113	1520.0007	1.6878	-122.3211
26	3	1.7267	3.4500	1063.9559	201.1105	22.8000	0.177299	1083.6789	1.6230	-112.2279
26	4	1.7267	3.6137	1017.7007	227.3063	25.5073	0.191703	1035.8071	1.5057	-100.9358

L	M	X (in)	Y (in)	U (fps)	V (fps)	P (psia)	RHO	Q (fps)	Mach	T (F)
26	5	1.7267	3.7735	1380.0098	207.8969	27.9910	0.200890	1395.9772	1.0830	-91.2585
26	6	1.7267	3.9333	1302.5008	187.8090	30.0071	0.217269	1355.5830	1.0228	-82.2083
27	1	1.9733	3.0909	1059.0269	257.2658	22.5791	0.175550	1081.5308	1.6221	-112.8006
27	2	1.9733	3.2505	1007.6522	280.2170	22.7877	0.176792	1075.2885	1.6135	-112.0919
27	3	1.9733	3.0101	1021.8776	293.0628	20.3002	0.185300	1051.7650	1.5730	-105.5272
27	4	1.9733	3.5696	1390.0753	279.9659	26.1367	0.195000	1421.9095	1.5251	-98.2955
27	5	1.9733	3.7292	1365.8397	260.7981	28.2135	0.206032	1390.5157	1.0750	-90.3801
27	6	1.9733	3.8868	1330.1585	200.7033	30.0107	0.217288	1355.7051	1.0229	-82.2369
28	1	2.2200	3.0975	1910.7995	-298.7612	25.9098	0.192169	1932.5583	1.5969	-103.1005
28	2	2.2200	3.2097	1906.2072	-302.8893	25.0096	0.189801	1938.9577	1.5561	-109.9101
28	3	2.2200	3.3620	1391.8996	-333.8577	25.5297	0.192227	1931.3259	1.5921	-101.5290
28	4	2.2200	3.5193	1370.9298	-339.0730	26.7231	0.198370	1910.5562	1.5090	-96.3879
28	5	2.2200	3.6766	1396.7931	-320.5292	28.5979	0.207899	1389.3600	1.4667	-89.2631
28	6	2.2200	3.8339	1317.7933	-293.3750	30.3985	0.217221	1350.0599	1.9170	-82.2726
29	1	2.9667	3.0090	1291.9779	-227.8106	32.9970	0.231216	1311.9087	1.3696	-75.3852
29	2	2.9667	3.1572	1303.0991	-283.3728	31.6862	0.229897	1333.5995	1.3950	-79.7109
29	3	2.9667	3.3109	1313.7529	-335.5893	30.9038	0.218006	1355.9372	1.4256	-83.5673
29	4	2.9667	3.4637	1313.3000	-362.5929	29.8082	0.219592	1362.9359	1.4352	-89.9830
29	5	2.9667	3.6169	1309.8925	-363.9159	30.1072	0.215866	1359.3710	1.4292	-83.5939
29	6	2.9667	3.7702	1315.1981	-339.5192	30.3952	0.217210	1358.3199	1.4257	-82.2996
30	1	2.7131	2.9605	1199.3369	-210.5938	90.0065	0.269795	1212.7615	1.2251	-52.1976
30	2	2.7133	3.1101	1210.9119	-238.5102	38.6329	0.258299	1239.1772	1.2530	-56.2885
30	3	2.7133	3.2598	1290.8253	-278.9910	36.0291	0.295565	1271.6627	1.3037	-69.0377
30	4	2.7133	3.4095	1267.3081	-319.9771	33.3000	0.232090	1305.8638	1.3535	-72.6991
30	5	2.7133	3.5591	1293.9769	-339.6986	31.3972	0.222919	1336.5995	1.3968	-78.9811
30	6	2.7133	3.7088	1306.3885	-325.0857	30.3957	0.217212	1346.2286	1.4100	-82.2909
31	1	2.9600	2.9170	1096.6960	-193.3771	47.0660	0.298373	1113.6193	1.1009	-39.2290
31	2	2.9600	3.0631	1118.7287	-193.6976	95.5797	0.291692	1135.3698	1.1278	-38.2301
31	3	2.9600	3.2092	1167.8982	-221.2927	91.6999	0.273125	1188.6785	1.1953	-98.9999
31	4	2.9600	3.3552	1221.3161	-267.3618	36.7919	0.299537	1250.2381	1.2785	-62.0356
31	5	2.9600	3.5013	1278.0603	-305.8819	32.6871	0.228973	1319.1593	1.3657	-79.6800
31	6	2.9600	3.6979	1297.5788	-322.8935	30.3963	0.217213	1337.1503	1.9035	-82.2872

Note: This chapter was reconstructed from OCR-extracted text and program code analysis. For production use, consult the original LASL technical documentation and verify input/output specifications with the actual Fortran source code listings in Appendix C.

References

1. L. M. Saunders, “Numerical Solution of the Flow Field in the Throat Region of a Nozzle,” Brown Engineering Co. report BSVD-P-66-TN-001 (NASA CR 82601), August 1966.
2. R. F. Cuffel, L. H. Back, and P. F. Massier, “Transonic Flow-Field in a Supersonic Nozzle with Small Throat Radius of Curvature,” *AIAA J.* 7, 1364–1366, July 1969.
3. A. A. Migdal, E. J. Wuchina, and W. D. Boyd, “Method for Predicting the Nozzle Operating Line,” NASA Lewis Research Center Report NASA CR-72547, April 1968.
4. S. Prozan, “Numerical Solution of Transonic Flow in a Nozzle,” Grumman Aircraft Engineering Corporation Report, 1966.
5. C. W. Laval, “Calculation of Two-Dimensional Compressible Flow in Convergent-Divergent Channels,” AEDC Report TR-69-120, August 1969.
6. P. Serra, “Numerical Simulation of the Transonic Flow in a Convergent-Divergent Nozzle,” CNES Report, 1970.
7. C. A. Brown and H. Ozcan, “Numerical Solution of Steady Flow in a Nozzle,” Douglas Aircraft Report, 1969.
8. R. W. MacCormack, “The Effect of Viscosity in Hypervelocity Impact Cratering,” AIAA Paper 69-354, 1969.
9. G. Moretti and M. Abbett, “A Time-Dependent Method for the Analysis of Transonic Flows,” *AIAA J.* 4, 2136–2141, December 1966.
10. G. Moretti, “Transonic and Supersonic Flow Computations,” *Advances in Applied Mathematics* 2, 1981.
11. J. von Neumann and R. D. Richtmyer, “A Method for the Numerical Calculation of Hydrodynamic Shocks,” *J. Applied Phys.* 21, 232–237, March 1950.
12. *Ibid*, Reference 11.
13. M. C. Cline, “NAP: A Computer Program for the Computation of Two-Dimensional, Time-Dependent, Inviscid Nozzle Flow,” Los Alamos Scientific Laboratory Report, 1977.

14. R. E. Loh and R. L. Loh, "A Comparison of Computer Execution Times," TRW Technical Report, 1975.
15. IBM Corporation, "System/360 Performance Characteristics," IBM Technical Manual, 1971.
16. R. H. Prozan and D. E. Kooker, "Numerical Analysis of Steady Nozzle Flow Using Relaxation Methods," AIAA Paper, 1971.
17. T. P. Thornock, "Experimental Study of Converging Nozzle Flow," NASA Technical Memorandum, June 1968.
18. D. L. Bresnahan and A. L. Johns, "Experimental Flow in a Plug Nozzle," NASA Contractor Report, 1972.

Appendix A

Characteristic Relations

A.1 $\eta = \text{constant}$ Reference Plane

[To be completed: Derivation of characteristic relations for inlet and exit boundaries (Appendix A from original)]

A.2 $\zeta = \text{constant}$ Reference Plane

[To be completed: Derivation of characteristic relations for wall and centerbody boundaries (Appendix B from original)]

Appendix B

Fortran Code Listing (LASL Identification: LP-0537)

B.1 Main Program (fortran_main.f)

This is the main program that orchestrates the NAP solver. It handles input/output, initialization, and time-stepping control. *OCR note: this listing received a conservative normalization pass for obvious token errors; unresolved ambiguities are tracked in conversion notes.*

```
1  PROGRAM MAIN(INPUT,OUTPUT,FILM,PUNCH,TAPE5=INPUT,TAPE6=OUTPUT,  
2  1TAPE7=FILM)  
3  C  
4  C *****  
5  C NAP, A COMPUTER PROGRAM FOR THE COMPUTATION OF TWO-DIMENSIONAL,  
6  C TIME-DEPENDENT, INVISCID NOZZLE FLOW  
7  C  
8  C BY MICHAEL C. CLINE, T-3  
9  C LOS ALAMOS SCIENTIFIC LABORATORY  
10 C  
11 C *****  
12 C  
13 C PROGRAM ABSTRACT  
14 C  
15 C THE EQUATIONS OF MOTION FOR TWO-DIMENSIONAL, TIME DEPENDENT,  
16 C INVISCID FLOW IN A NOZZLE ARE SOLVED USING THE SECOND-ORDER,  
17 C MACCORMACK FINITE-DIFFERENCE SCHEME, THE FLUID IS ASSUMED TO BE  
18 C A PERFECT GAS, ALL BOUNDARY CONDITIONS ARE COMPUTED USING A  
19 C SECOND-ORDER, REFERENCE PLANE CHARACTERISTIC SCHEME, THE STEADY  
20 C STATE SOLUTION IS OBTAINED AS THE ASYMPTOTIC SOLUTION FOR LARGE  
21 C TIME, THE NOZZLES MAY BE EITHER CONVERGING, CONVERGING-DIVERGING,  
22 C OR PLUG GEOMETRIES.  
23 C  
24 C DIMENSION TITLE(8), UI(21), VI(21), PI(21), ROI(21)  
25 C COMMON /AV/ IAV,CAV,NST,SMP,LSS,CTA,XMU,XLA,RKMU,QUT(81,21),QVT(81  
26 1,21),QPT(81,21)  
27 C COMMON /ONESID/ UD(4),VD(4),PD(4),ROD(4)  
28 C COMMON /SOLUTIN/ U(81,21,2),V(81,21,2),P(81,21,2),RO(81,21,2)  
29 C COMMON /CNTRL/ LMAX,MMAX,NMAX,NPRINT,TCONV,FDT,GAMMA,ARGAS,GAMI,GA  
30 1M2,L1,L2,L3,M1,M2,DX,DY,DT,N,N1,N3,NASM,IVEL,ICHR,N1D,LJET,JFLAG,  
31 2IERR,IUI,IUO,DXR,DYR,LD,MD,LMD1,LMD3,IB,RSTAR,RSTARS,NPLOT,G,PC,TC  
32 3,LC,PLOW,ROLOW  
33 C COMMON /GEMTRY/ NGEOM,XI,RI,XT,RT,XE,RE,RCI,RCT,ANGL,ANGE,XW(81),  
34 1YW(81),XWI(81),YWI(81),NXNY(81),NWPTS,IINT,IDIF,LT,NDIM  
35 C COMMON /GCB/ NGCB,XICB,RICB,XTCB,RTCB,XECB,RECB,RCICB,RCTCB,ANGICB  
36 1,ANGEGB,XCB(81),YCB(81),XCB1(81),YCB1(81),NXNYCB(81),NCBPTS,IINTCB  
37 2,IDIFCB,LECB  
38 C COMMON /BOC/ PT(21),TT(21),THETA(21),PE,MASSE,MASSI,MASST,THRUST,N  
39 3STAG
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40      REAL, MN3,NXNY,MASSI,MASST,NXNYCB,MASSE
41      NAMELIST /CNTRL/ LMAX,MMAX,NMAX,NPRINT,TCONV,FDT,TSTOP,GAMMA,RGAS,
42      1NASM,NAME,NCONVI,NST,IUI,IUO,SMP,IPUNCH,IAV,CAV,NPLOT,IEX,LSS,CTA,
43      2XMU,XLA,RKMU,IUNIT,FLOW,ROLOW
44      NAMELIST /IVS/ U,V,P,RO,N1D,NSTART,TSTART,RSTAR,RSTARS
45      NAMELIST /GEMIRY/ NDIM,XI,RI,RT,XE,RCI,RCT,ANGI,ANGE,NGEOM,XWI,YWI
46      1,NWPTS,IINT,IDIF,LJET,JFLAG,NXNY,YW
47      NAMELIST /GCBL/ NGCB,RICB,RTCB,RCICB,RCTCB,ANGICB,ANGEGB,YCB,NXNYC
48      1B,XCBI,YCBI,NCBPTS,IINTCB,IDIFCB
49      NAMELIST /BC/ PT,TT,THETA,PE,NSTAG,ISUPER,UI,VI,PI,ROI
50 C
51 C      READ IN DATA
52 C
53 10 TCONV=0.0 $ FDT=1.0 $ TSTOP=1.0 $ NASM=1 $ NSTAG=0 $ NAME=0
54 IPUNCH=0 $ NGCB=0 $ IINTCB=1 $ IDIFCB=1 $ NSTART=0 $ TSTART=0.0
55 IINT=1 $ IDIF=1 $ NMAX=0 $ NPRINT=0 $ GAMMA=1.4 $ RGAS=53.35
56 N1D=1 $ NDIM=1 $ THETA(1)=0.0 $ PE=14.7 $ NST=0 $ N=0 $ IEX=1
57 NCONVI=1 $ IERR=0 $ JFLAG=0 $ IUI=1 $ IUO=1 $ SMP=0.95 $ ISUPER=0
58 IAV=0 $ CAV=4.0 $ NPLOT=-1 $ G=32.174 $ PC=144.0 $ TC=460.0
59 LC=12.0 $ IUNIT=0 $ LSS=2 $ CTA=0.5 $ XMU=0.2 $ XLA=1.0
60 RKMU=0.7 $ FLOW=0.01 $ ROLOW=0.0001 $ RSTAR=0.0 $ RSTARS=0.0
61 READ 650, TITLE
62 IF (EOF,5) 20,30
63 20 STOP
64 30 READ (5,CNTRL)
65 READ (5,IVS)
66 READ (5,GEMIRY)
67 READ (5,GCBL)
68 READ (5,BC)
69 IF (NAME,EQ,0) GO TO 40
70 WRITE (6,CNTRL)
71 WRITE (6,IVS)
72 WRITE (6,GEMIRY)
73 WRITE (6,GCBL)
74 WRITE (6,BC)
75 C
76 C      PRINT INPUT DATA
77 C
78 40 PRINT 660
79 PRINT 690
80 PRINT 680
81 PRINT 700
82 PRINT 670
83 PRINT 710, TITLE
84 PRINT 670
85 PRINT 720
86 NPRINT=ABS(FLOAT(NPRINT))
87 PRINT 730,
88      LMAX,MMAX,NMAX,NPRINT,TCONV,FDT,NSTAG,NASM,IUNIT,IUI,IUO,IEX,NCONVI,TSTOP,N1D,NPLOT,IPUNCH,ISUPER,IAV,CAV,XMU,XLA,RKMU,CTA,LSS,SMP,
89 PRINT 670
90 IF (IUI,EQ,1) PRINT 740, GAMMA,RGAS
91 IF (IUI,EQ,2) PRINT 750, GAMMA,RGAS
92 PRINT 670
93 PRINT 780
94 IF (NDIM,EQ,0) PRINT 790
95 IF (NDIM,EQ,1) PRINT 800
96 C
97 C      CALCULATE THE NOZZLE RADIUS AND NORMAL
98 C
99      PRINT 670
100      CALL GEOM
101      IF (IERR,NE,0) GO TO 10
102      DY=1.0/FLOAT(MMAX-1)
103      IF (NGCB,NE,0) GO TO 60
104      RICB=0.0
105      RTCB=0.0
106      DO 50 L=1,IMAX
107      YCB(L)=0.0
108      NXNYCB(L)=0.0
109 50 CONTINUE
110 GO TO 90
111 60 XICB=XI
112 XECB=XE
113 CALL GEOMCB
114 LT=1 $ XI=XICB $ XE=XECB
115 Y0=0.0

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115 DO 80 L=1,LMAX
116 IF (NDIM,EQ,0) Y=YW(L)-YCB(L)
117 IF (NDIM,EQ,1) Y=YW(L)**2-YCB(L)**2
118 IF (Y,GT,0.0) GO TO 70
119 PRINT 920
120 GO TO 10
121 70 IF (Y,LT,Y0) LT=L
122 Y0=Y
123 80 CONTINUE
124 90 IF (NSTAG,NE,0) GO TO 110
125 DO 100 M=2,MMAX
126 PT(M)=PT(1)
127 TT(M)=TT(1)
128 THETA(M)=THETA(1)
129 100 CONTINUE
130 PRINT 670
131 IF (IUI,EQ,1) PRINT 760, PT(1),TT(1),THETA(1),PE
132 IF (IUI,EQ,2) PRINT 770, PT(1),TT(1),THETA(1),PE
133 GO TO 130
134 110 PRINT 660
135 IF (IUI,EQ,1) PRINT 890, PE
136 IF (IUI,EQ,2) PRINT 770, PE
137 DO 120 M=1,MMAX
138 PRINT 910, M,PT(M),TT(M),THETA(M)
139 120 CONTINUE
140 C
141 C CONVERT METRIC UNITS TO ENGLISH UNITS
142 C
143 130 IF (IUI,EQ,1) GO TO 180
144 RSTAR=RSTAR/2.54
145 RSTARS=RSTARS/6.4516
146 RGAS=RGAS/5.38032
147 DO 140 M=1,MMAX
148 PT(M)=PT(M)/6.8948
149 TT(M)=(TT(M)+40.0)*9.0/5.0-40.0
150 140 CONTINUE
151 PE=PE/6.8948
152 IF (ISUPER,EQ,0) GO TO 160
153 DO 150 M=1,MMAX
154 UI(M)=UI(M)/0.3048
155 VI(M)=VI(M)/0.3048
156 PI(M)=PI(M)/6.8948
157 ROI(M)=ROI(M)/16.02
158 150 CONTINUE
159 160 IF (NID,NE,0) GO TO 180
160 IF (NSTART,NE,0) GO TO 180
161 DO 170 L=1,LMAX
162 DO 170 M=1,MMAX
163 U(L,M,1)=U(L,M,1)/0.3048
164 V(L,M,1)=V(L,M,1)/0.3048
165 P(L,M,1)=P(L,M,1)/6.8948
166 RO(L,M,1)=RO(L,M,1)/16.02
167 170 CONTINUE
168 C
169 C CONVERT INPUT DATA UNITS TO INTERNAL UNITS
170 C
171 180 IF (IUNIT,EQ,0) GO TO 190
172 PC=1.0 $ LC=1.0 $ G=1.0
173 TC=0.0
174 190 TCONV=TCONV/100.0
175 T=ISTART*LC
176 TSTOP=TSTOP*LC
177 DO 200 L=1,LMAX
178 XWI(L)=0.0
179 200 CONTINUE
180 DO 210 M=1,MMAX
181 PT(M)=PT(M)*PC
182 TT(M)=TT(M)+TC
183 THETA(M)=THETA(M)*0.0174533
184 210 CONTINUE
185 PE=PE*PC
186 IF (NID,NE,0) GO TO 230
187 DO 220 L=1,LMAX
188 DO 220 M=1,MMAX
189 P(L,M,1)=P(L,M,1)*PC
190 RO(L,M,1)=RO(L,M,1)/G

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191 220 CONTINUE
192 230 GAM1=GAMMA/(GAMMA-1.0)
193     GAM2=(GAMMA-1.0)/2.0
194     IF (ISUPER,EQ,0) GO TO 250
195     DO 240 M=1,MMAX
196     U(1,M,1)=UI(M)
197     V(1,M,1)=VI(M)
198     P(1,M,1)=PI(M)*PC
199     RO(1,M,1)=ROI(M)/G
200     U(1,M,2)=U(1,M,1)
201     V(1,M,2)=V(1,M,1)
202     P(1,M,2)=P(1,M,1)
203     RO(1,M,2)=RO(1,M,1)
204 240 CONTINUE
205 250 L1=LMAX-1
206     L2=LMAX-2
207     L3=LMAX-3
208     M1=MMAX-1
209     M2=MMAX-1
210     IF (NID,EQ,0) GO TO 260
211 C
212 C COMPUTE THE 1-D INITIAL-DATA SURFACE
213 C
214     CALL ONEDIM
215     IF (IERR,NE,0) GO TO 10
216 C
217 C COMPUTE THE INITIAL-DATA SURFACE MASS FLOW AND THRUST
218 C
219 260 IF (NPRINT,GT,0) GO TO 270
220     NPRINT=-NPRINT
221     GO TO 340
222 270 CALL MASFLO (0)
223 C
224 C CALCULATE AND PRINT THE INITIAL-VALUE SURFACE
225 C
226     DO 330 IU=1,2
227     IF (IUO,EQ,1,AND,IU,EQ,2) GO TO 330
228     IF (IUO,EQ,2,AND,IU,EQ,1) GO TO 330
229     NLINE=0
230     PRINT 660
231     PRINT 810, TSTART,NSTART
232     PRINT 820
233     IF (IU,EQ,1) PRINT 830
234     IF (IU,EQ,2) PRINT 840
235     PRINT 670
236     X=XI-DX
237     DO 300 L=1,LMAX
238     X=X+DX
239     CALL MAP (0,L,1,AL,BE,DE,LD1,AL1,BE1,DE1)
240     DYIO=DY/BE
241     Y=YCB(L)-DYIO
242     DO 300 M=1,MMAX
243     Y=Y+DYIO
244     VELMAG=SQRT(U(L,M,1)**2+V(L,M,1)**2)
245     XMACH=VELMAG/SQRT(GAMMA*P(L,M,1)/RO(L,M,1))
246     PRES=P(L,M,1)/PC
247     RHO=RO(L,M,1)/G
248     TEMP=P(L,M,1)/RHO/RGAS-TC
249     XP=X
250     YP=Y
251     UP=U(L,M,1)
252     VP=V(L,M,1)
253     IF (IU,EQ,1) GO TO 280
254     XP=XP*2.54
255     YP=YP*2.54
256     UP=UP*0.3048
257     VP=VP*0.3048
258     PRES=PRES*6.8948
259     RHO=RHO*16.02
260     VELMAG=VELMAG*0.3048
261     TEMP=(TEMP+40.0)*5.0/9.0-40.0
262 280 NLINE=NLINE+1
263     IF (NLINE,LT,55) GO TO 290
264     PRINT 660
265     PRINT 810, TSTART,NSTART
266     PRINT 820

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267 IF (IU,EQ,1) PRINT 830
268 IF (IU,EQ,2) PRINT 840
269 PRINT 670
270 NLINE=1
271 290 PRINT 850, L,M,XP,YP,UP,VP,PRES,RHO,VELMAC,XMACH,TEMP
272 300 CONTINUE
273 IF (IU,EQ,2) GO TO 310
274 PRINT 870, MASST,THRUST,MASSI,MASSE
275 GO TO 320
276 310 MASST=MASST*0.4536
277 MASSI=MASSI*0.4536
278 MASSE=MASSE*0.4536
279 THRUST=THRUST*4.4477
280 PRINT 880, MASST,THRUST,MASSI,MASSE
281 320 IF (IUO,NE,3) GO TO 340
282 330 CONTINUE
283 340 IF (NPLOT,LE,0) GO TO 350
284 CALL PLOT (TITLE,TSTART,NSTART)
285 PRINT 1030, NSTART
286 350 IF (NMAX,EQ,0) GO TO 10
287 C
288 C INITIALIZE THE TIME STFP INTEGRATION LOOP PARAMETERS
289 C
290 N1=1 $ N3=2 $ DQM=0.0 $ NS=0 $ NCONV=0 $ NC=0 $ LDUM=1 $ NPC=0
291 DXR=1.0/DX $ DYR=1.0/DY $ DXRS=DXR*DXR $ DYRS=DYR*DYR
292 LD=81 $ MD=21 $ LMD=LD*MD
293 IF (NASM,NE,0,AND,LT,NE,1) LDUM=LT-1
294 NPD=0
295 IF (JFLAG,EQ,0) GO TO 360
296 UD(1)=U(LJET-1,MMAX,N1)
297 VD(1)=V(LJET-1,MMAX,N1)
298 PD(1)=P(LJET-1,MMAX,N1)
299 ROD(1)=RO(LJET-1,MMAX,N1)
300 UD(2)=UD(1)
301 VD(2)=VD(1)
302 PD(2)=PD(1)
303 ROD(2)=ROD(1)
304 C
305 C ENTER THE TIME STEP INTERGRATION LOOP
306 C
307 360 DO 580 N=1,NMAX
308 NDP=NPD+1
309 IF (NPD,NE,10) GO TO 370
310 NP=N+NSTART
311 PRINT 1040, NP
312 NPD=0
313 370 CONTINUE
314 LMD1=LMD*(N1-1)
315 LMD3=LMD*(N3-1)
316 C
317 C CALCULATE DELTA T
318 C
319 DO 380 L=1,IMAX
320 CALL MAP (0,L,MD,AL,BE,DE,LD1,AL1,BE1,DE1)
321 DXDY=DXRS*BE*BE*DYRS
322 DO 380 M=1,MMAX
323 LMN1=L+LD*(M-1)+LMD1
324 QS=U(LMN1)*U(LMN1)+V(LMN1)*V(LMN1)
325 AS=GAMMA*P(LMN1)/RO(LMN1)
326 UPA=SQRT(QS*DXDY)+SQRT(AS*DXDY)
327 IF (L,EQ,1,AND,M,EQ,1) UPAM=UPA
328 IF (UPA,GT,UPAM) UPAM=UPA
329 380 CONTINUE
330 DT=FDT/UPAM
331 T=T+DT
332 IF (T,LE,TSTOP) GO TO 390
333 T=T-DT
334 DT=TSTOP-T
335 T=TSTOP
336 C
337 C DETERMINE IF THE EXIT FLOW IS SUBSONIC OR SUPERSONIC
338 C
339 390 IVEL=0
340 IF (QS,GE,AS) IVEL=1
341 C
342 c CALCULATE THE NOZZLE WALL AND INTERIOR MESH POINTS

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343 C
344     IF (IAV,NE,0) CALL SHOCK (1)
345     ICHAR=1
346     IB=1
347     CALL INTER
348     CALL WALL
349     IF (IERR,NE,0) GO TO 10
350     IF (NGCB,EQ,0) GO TO 400
351     IB=2
352     CALL WALL
353     IF (IERR,NE,0) GO TO 10
354 400 ICHAR=2
355     IB=1
356     CALL INTER
357     CALL WALL
358     IF (IERR,NE,0) GO TO 10
359     IF (NGCB,EQ,0) GO TO 410
360     IB=2
361     CALL WALL
362     IF (IERR,NE,0) GO TO 10
363 C
364 C     EXTRAPOLATE THE EXIT MESH POINTS FOR SUPERSONIC FLOW
365 C
366 410 DO 420 M=1,MMAX
367     U(LMAX,M,N3)=U(L1,M,N3)+IEX*(U(L1,M,N3)-U(L2,M,N3))
368     V(LMAX,M,N3)=V(L1,M,N3)+IEX*(V(L1,M,N3)-V(L2,M,N3))
369     P(LMAX,M,N3)=P(L1,M,N3)+IEX*(P(L1,M,N3)-P(L2,M,N3))
370     RO(LMAX,M,N3)=RO(L1,M,N3)+IEX*(RO(L1,M,N3)-RO(L2,M,N3))
371     IF (P(LMAX,M,N3),GT,0.0,AND,RO(LMAX,M,N3),GT,0.0) GO TO 420
372     P(LMAX,M,N3)=P(L1,M,N3)
373     RO(LMAX,M,N3)=RO(L1,M,N3)
374 420 CONTINUE
375     V(LMAX,MMAX,N3)=-U(LMAX,MMAX,N3)*NXNY(LMAX)
376     V(LMAX,1,N3)=-U(LMAX,1,N3)*NXNYCB(LMAX)
377 C
378 C     CALCULATE THE NOZZLE INLET MESH POINTS
379 C
380     IF (ISUPER,EQ,0) CALL INLET
381 C
382 C     CALCULATE THE NOZZLE EXIT MESH POINTS FOR SUBSONIC FLOW
383 C
384     IF (IVEL,EQ,0) CALL EXITT
385     IF (N,LE,NST) CALL SHOCK (2)
386 C
387 C     DETERMINE THE MAXIMUM (DELTA U)/U
388 C
389     IF (TCONV,LE,0.0) GO TO 440
390     DDQM=0.0
391     DO 430 L=LDUM,LMAX
392     DO 430 M=1,MMAX
393     IF (U(L,M,N1),EQ,0.0) GO TO 430
394     DQ=ABS((U(L,M,N3)-U(L,M,N1))/U(L,M,N1))
395     IF (DQ,GT,DDQM) DDQ=DQ
396 430 CONTINUE
397 440 NC=NC+1
398     NPC=NPC+1
399     IF (DDQ,GE,TCONV) GO TO 450
400     NCONV=NCONV+1
401     IF (NCONV,EQ,1) NCHECK=N-1
402     IF (NCONV,GE,NCONV) NC=NPRINT
403 450 IF (N,EQ,NMAX) NC=NPRINT
404     IF (N,GE,NCHECK+NCONV) NCONV=0
405     IF (T,EQ,TSTOP) NC=NPRINT
406     IF (NC,EQ,NPRINT) GO TO 460
407     IF (NPC,EQ,NPLOT) GO TO 550
408     GO TO 570
409 C
410 C     COMPUTE THE SOLUTION SURFACE MASS FLOW AND THRUST
411 C
412 460 ICN=0
413     IF (JFLAG,EQ,0) GO TO 470
414     IF (LT,NE,LJET-1) GO TO 470
415     UDUM=U(LT,MMAX,N3)
416     RODUM=RO(LT,MMAX,N3)
417     U(LT,MMAX,N3)=UD(3)
418     RO(LT,MMAX,N3)=ROD(3)

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419      ICN=1
420 470    CALL MASFLO (1)
421      IF (ICN,EQ,0) GO TO 480
422      U(LT,MMA,N3)=UDUM
423      RO(LT,MMA,N3)=RODUM
424  C
425  C      CALCULATE AND PRINT THE SOLUTION SURFACE
426  C
427 480    DO 540 IU=1,2
428      IF (IUO,EQ,1,AND,IU,EQ,2) GO TO 540
429      IF (IUO,EQ,2,AND,IU,EQ,1) GO TO 540
430      NLINE=0
431      PRINT 660
432      TIME=T/LC
433      DTIME=DT/LC
434      NP=N+NSTART
435      PRINT 860, NP,TIME,DTIME
436      PRINT 820
437      IF (IU,EQ,1) PRINT 830
438      IF (IU,EQ,2) PRINT 840
439      PRINT 670
440      X=XI-DX
441      DO 510 L=1,LMAX
442      X=X+DX
443      CALL MAP (0,L,1,AL,BE,DE,LD1,AL1,BE1,DE1)
444      DYIO=DY/BE
445      Y=YCB(L)-DYIO
446      DO 510 M=1,MMA
447      Y=Y+DYIO
448      VELMAG=SQRT(U(L,M,N3)**2+V(L,M,N3)**2)
449      XMACH=VELMAG/SQRT(GAMMA*P(L,M,N3)/RO(L,M,N3))
450      PRES=P(L,M,N3)/PC
451      RHO=RO(L,M,N3)/G
452      TEMP=P(L,M,N3)/RHO/RGAS-TC
453      XP=X
454      YP=Y
455      UP=U(L,M,N3)
456      VP=V(L,M,N3)
457      IF (IU,EQ,1) GO TO 490
458      XP=XP*2.54
459      YP=YP*2.54
460      UP=UP*0.3048
461      VP=VP*0.3048
462      PRES=PRES*6.8948
463      RHO=RHO*16.02
464      VELMAG=VELMAG*0.3048
465      TEMP=(TEMP+40.0)*5.0/9.0-40.0
466 490    NLINE=NLINE+1
467      IF (NLINE,LT,55) GO TO 500
468      PRINT 660
469      PRINT 860, NP,TIME,DTIME
470      PRINT 820
471      IF (IU,EQ,1) PRINT 830
472      IF (IU,EQ,2) PRINT 840
473      PRINT 670
474      NLINE=1
475 500    PRINT 850, L,M,XP,YP,UP,VP,PRES,RHO,VELMAG,XMACH,TEMP
476 510    CONTINUE
477      IF (IU,EQ,2) GO TO 520
478      PRINT 870, MASST,THRUST,MASSI,MASSE
479      GO TO 530
480 520    MASST=MASST*0.4536
481      MASSI=MASSI*0.4536
482      MASSE=MASSE*0.4536
483      THRUST=THRUST*4.4477
484      PRINT 880, MASST,THRUST,MASSI,MASSE
485 530    IF (IUO,NE,3) GO TO 550
486 540    CONTINUE
487 550    IF (NPLOT,LT,0) GO TO 560
488      TIME=T/LC $ NP=N+NSTART
489      CALL PLOT (TITLE,TIME,NP)
490      PRINT 1030, NP
491  C
492  C      CHECK FOR CONVERGENCE OF THE STEADY STATE SOLUTION
493  C
494 560    IF (DQM,LT,TCONV) GO TO 590

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495      IF (T,EQ,TSTOP) GO TO 590
496      IF (N,EQ,NMAX) GO TO 590
497      IF (NC,EQ,NPRINT) NC=0
498      IF (NPC,EQ,NPLOT) NPC=0
499 570    CONTINUE
500      NNN=N1
501      N1=N3
502      N3=NNN
503 580    CONTINUE
504      C
505      C      PUNCH A SIVS NAMELIST FOR RESTART
506      C
507 590      IF (NPLOT,GE,0) CALL ADV (10)
508      IF (IPUNCH,EQ,0) GO TO 10
509      DO 600 L=1,LMAX
510      DO 600 M=1,MMAX
511      P(L,M,N3)=P(L,M,N3)/PC
512      RO(L,M,N3)=RO(L,M,N3)*G
513 600    CONTINUE
514      PUNCH 930, NP,TIME
515      DO 610 M=1,MMAX
516      PUNCH 940, M
517      PUNCH 950, (U(L,M,N3),L=1,LMAX)
518 610    CONTINUE
519      DO 620 M=1,MMAX
520      PUNCH 960, M
521      PUNCH 950, (V(L,M,N3),L=1,LMAX)
522 620    CONTINUE
523      DO 630 M=1,MMAX
524      PUNCH 970, M
525      PUNCH 980, (P(L,M,N3),L=1,LMAX)
526 630    CONTINUE
527      DO 640 M=1,MMAX
528      PUNCH 990, M
529      PUNCH 1000, (RO(L,M,N3),L=1,LMAX)
530 640    CONTINUE
531      PUNCH 1010
532      NCARDS=(LMAX/7+2)*MMAX*4+22)
533      PRINT 1020, NCARDS
534      GO TO 10
535      C
536      C      FORMAT STATEMENTS
537      C
538 650      FORMAT (8A10)
539 660      FORMAT (1H1)
540 670      FORMAT (1H )
541 680      FORMAT (1H0)
542 690      FORMAT (1H0,15X,NAP, A COMPUTER PROGRAM FOR THE COMPUTATION OF TWO-DIMENSIONAL, TIME-DEPENDENT, INVISCID NOZZLE
FLOW,/,/ ,37X,59HBY MICHAEL C. CLINE, T-3 - LOS ALAMOS SCIENTIFIC LABORATORY)
543 700      FORMAT (1H0,10X,18HPROGRAM ABSTRACT 26X,86HTHE EQUATIONS OF MOTION FOR TWO-DIMENSIONAL, TIME DEPENDENT,
INVISCID FLOW IN A NOZZLE,/,/ ,21X,93HARE SOLVED USING THE SECOND-ORDER, MACCORMACK FINITE-DIFFERENCE SCHEME, THE
FLUID IS ASSUMED,/,/ ,21X,95HTO BE A PERFECT GAS, ALL BOUNDARY CONDITIONS ARE COMPUTED USING A SECOND-ORDER,
REFERENCE PLANE,/,/ ,21X,91HCHARACTERISTIC SCHEME, THE STEADY STATE SOLUTION IS OBTAINED AS THE ASYMPTOTIC SOLUTION
FOR,/,/ ,21X,91HLARGE TIME, THE NOZZLES MAY BE EITHER CONVERGING, CONVERGING-DIVERGING, OR PLUG GEOMETRIES,/)
544 710      FORMAT (1H0,10X,11HJOB TITLE -//21X,8A10)
545 720      FORMAT (1H0,10X,20HCONTROL PARAMETERS -)
546 730      FORMAT
(1H0,20X,5HIMAX=,I2,2X,5HIMAX=,I2,3X,5HINMAX=,I4,2X,7HNPRINT=,I4,2X,6HITCONV=,F6.3,3X,4HFDT=,F4.2,2X,6HNSTAG=,I1,5X,5HINASM=,I1,4X,6HIUNTY
547 740      FORMAT (1H0,10X,13HFLUID MODEL - ,// ,21X,36HTHE RATIO OF SPECIFIC HEATS, GAMMA =,F6.4,26H AND THE GAS CONSTANT,
R =,F9.4,15H (FT-LBF/LBM-R))
548 750      FORMAT (1H0,10X,13HFLUID MODEL - ,// ,21X,36HTHE RATIO OF SPECIFIC HEATS, GAMMA =,F6.4,26H AND THE GAS CONSTANT,
R =,F9.4,9H (J/KG-K))
549 760      FORMAT (1H0,10X,21HBOUNDARY CONDITIONS - ,// ,21X,3HPT=,F9.4,7H (PSIA),5X,3HTT=,F9.4,4H (F),5X,6HIHETA=,F9.4,6H
(DEG),5X,3HPE=,F9.4,7H (PSIA))
550 770      FORMAT (1H0,10X,21HBOUNDARY CONDITIONS - ,// ,21X,3HPT=,F9.4,6H (KPA),5X,3HTT=,F9.4,4H (C),5X,6HIHETA=,F9.4,6H
(DEG),5X,3HPE=,F9.4,6H (KPA))
551 780      FORMAT (1H0,10X,15HFLOW GEOMETRY -)
552 790      FORMAT (1H0,20X,47HTWO-DIMENSIONAL, PLANAR FLOW HAS BEEN SPECIFIED)
553 800      FORMAT (1H0,20X,36HAXISYMMETRIC FLOW HAS BEEN SPECIFIED)
554 810      FORMAT (1H ,30HINITIAL-DATA SURFACE - TIME = ,F10.8,8H SECONDS,4H (N=,I4,1H))
555 820      FORMAT (1H0,11X,1HL,4X,1HM,9X,1HX,10X,1HY,10X,1HU,1IX,1HV,12X,1HP,11X,3HRHO,9X,1HQ,11X,4HMACH,8X,1HT)
556 830      FORMAT (1H ,25X,4H(IN),7X,4H(IN),6X,5H(FPS),7X,5H(FPS),7X,6H(PSIA),6X,9H(LBM/FT3),4X,5H(FPS),10X,2HNO,8X,3H(F))
557 840      FORMAT (1H ,25X,4H(CM),7X,4H(CM),6X,5H(MPS),7X,5H(MPS),7X,6H (KPA),7X,7H(KG/M3),5X,5H(MPS),10X,2HNO,8X,3H(C))
558 850      FORMAT (1H ,7X,215 ,4F12.4,F13.4,F12.6,3F12.4)
559 860      FORMAT (1H ,20HSOLUTION SURFACE NO.,I5,3H - ,7HTIME = ,F10.8,20H SECONDS (DELTA T = ,F10.8,1H))
560 870      FORMAT (1H0,10X,5HMASS=,F9.4,10H (LBM/SEC),5X,7HITHRUST=,F11.4,6H (LBF),5X,6HMASSI=,F9.4,5X,6HMASSE=,F9.4)

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561 880   FORMAT (1H0,10X,5HMASS=,F9.4,9H (KS/SEC) ,5X,7HTHRUST=,F11.4,10H (NEWTONS) ,5X,6HMASSI=,F9.4,5X,6HMASSE=,F9.4)
562 890   FORMAT (1H0,10X,21HBOUNDARY CONDITIONS
- ,//,22X,1HM,11X,8HPT(PSLA) ,10X,5HTT(F) ,10X,10HIHETA(DEC) ,10X,3HPE=,F7.3,7H (PSIA) ,/)
563 900   FORMAT (1H0,10X,21HBOUNDARY CONDITIONS
- ,//,22X,1HM,11X,7HPT(KPA) ,12X,5HTT(C) ,10X,10HIHETA(DEC) ,10X,3HPE=,F7.3,6H (KPA) ,/)
564 910   FORMAT (1H ,20X,I2,10X,F7.2,10X,F7.2,10X,F7.2)
565 920   FORMAT (1H0,78H***** THE RADIUS OF THE CENTERBOOY IS LARGER THAN THE NOZZLE WALL RADIUS *****
566 930   FORMAT (1X,18HSIVS NID=0 ,NSTART=,I4,8H,TSTART=,F14.10,1H,)
567 940   FORMAT (1X,4HU(1,,I2,5H,1) =)
568 950   FORMAT (1X,7 (F10.3,1H,))
569 960   FORMAT (1X,4HV(1,,I2,5H,1) =)
570 970   FORMAT (1X,4HP(1,,I2,5H,1) =)
571 980   FORMAT (1X,7 (F10.4,1H,))
572 990   FORMAT (1X,4HRHO(1,,I2,5H,1) =)
573 1000  FORMAT (1X,7 (F10.6,1H,))
574 1010  FORMAT (1X,1HS)
575 1020  FORMAT (1H0,27H***** EXPECT APPROXIMATELY ,I4,20H PUNCHED CARDS *****
576 1030  FORMAT (1H0,31H***** EXPECT FILM OUTPUT FOR N=,I4,6H *****
577 1040  FORMAT (1H ,2HN=,I4)
578     END

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B.2 Geometry Subroutine (geom.f)

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1  SUBROUTINE GEOM GEO 10
2  C GEO 20
3  C ***** GEO 30
4  C GEO 40
5  C THIS SUBROUTINE CALCULATES THE NOZZLE RADIUS AND OUTER NORMAL GEO 50
6  C GEO 60
7  C ***** GEO 70
8  C GEO 80
9  COMMON /AV/ IAV,CAV,NST,SMP,LSS,CTA,XMU,XLA,RKMU,QUT(81,21) ,QVT(81,GEO 90
10 1,21) ,QPT(81,21) GEO 100
11 COMMON /ONESID/ UD(4) ,VD(4) ,PD(4) ,ROD(4) GEO 110
12 COMMON /SOLUTN/ U(81,21,2) ,V(81,21,2) ,P(81,21,2) ,RO(81,21,2) GEO 120
13 COMMON /CNTRLC/ LMAX,MMAX,NMAX,NPRINT,TCONV,FDT,GAMMA,RGAS,GAMI,GAGED GEO 130
14 1M2,L1,L2,L3,M1,M2,DX,DY,DT,N,N1,N3,NASM,IVEL,ICHAR,NID,LJET,JFLAG,GEO 140
15 2IERR,IUI,IUO,DXR,DYR,LD,MD,LMD1,LMD3,IB,RSTAR,RSTARS,NPLOT,G,PC,TCGEO 150
16 3,LC,PLOW,ROLOW GEO 160
17 COMMON /GEMIRYC/ NGEOM,XI,RI,XT,RT,XE,RE,RCI,RCT,ANGI,ANGE,XW(81) ,GEO 170
18 YW(81) ,XWI(81) ,YWI(81) ,NXNY(81) ,NWPTS,IINT,IDIF,LT,NDIM GEO 180
19 COMMON /GCB/ NGCB,XICB,RICB,XTCB,RTCB,XECB,RECB,RCICB,RCTCB,ANGICB,GEO 190
20 2,ANGECE,XCB(81) ,YCB(81) ,XCBI(81) ,YCB(81) ,NXNYCB(81) ,NCBPTS,IINTCB,GEO 200
21 3,IDIFCB,LECB GEO 210
22 COMMON /BOC/ PT(21) ,TT(21) ,THETA(21) ,PE,MASSE,MASSI,MASST,THRUST,NGEO 220
23 1STAG GEO 230
24 REAL MN3,NXNY,MASSI,MASST,NXNYCB,MASSE GEO 240
25 C GEO 250
26 GO TO (10,30,120,170) , NGEOM GEO 260
27 C GEO 270
28 C CONSTANT AREA DUCT CASE GEO 280
29 C GEO 290
30 10 PRINT 230 GEO 300
31 IF (IUI,EQ,1) PRINT 250 , XI,RI,XE GEO 310
32 IF (IUI,EQ,2) PRINT 260 , XI,RI,XE GEO 320
33 LT=LMAX GEO 330
34 DX=(XE-XI) / (LMAX-1) GEO 340
35 XT=XE GEO 350
36 RT=RI GEO 360
37 RE=RI GEO 370
38 DO 20 L=1 ,LMAX GEO 380
39 YW(L)=RI GEO 390
40 NXNY(L)=0.0 GEO 400
41 20 CONTINUE GEO 410
42 IF (JFLAG,EQ,0) GO TO 210 GEO 420
43 C GEO 430
44 XWI=XI+(LJET-2)*DX GEO 440
45 IF (IUI,EQ,1) PRINT 370 , XWI,LJET,LMAX GEO 450
46 IF (IUI,EQ,2) PRINT 380 , XWI,LJET,LMAX GEO 460
47 GO TO 210 GEO 470
48 C GEO 480
49 C CIRCULAR-ARC, CONICAL NOZZLE CASE GEO 490

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50	C		GEO	500
51	30	PRINT 230	GEO	510
52		IF (RCI,EQ,0.0,OR,RCT,EQ,0.0) GO TO 200	GEO	520
53		ANI=ANGI*3.141593/180.0	GEO	530
54		ANE=ANGE*3.141593/180.0	GEO	540
55		XTAN=XI+RCI*SIN(ANI)	GEO	550
56		RTAN=RI+RCI*(COS(ANI)-1.0)	GEO	560
57		RT1=RT-RCT*(COS(ANI)-1.0)	GEO	570
58		XT1=XTAN+(RTAN-RT1)/TAN(ANI)	GEO	580
59		IF (XT1,GE,XTAN) GO TO 40	GEO	590
60		XT1=XTAN	GEO	600
61		RT1=RTAN	GEO	610
62	40	XT=XT1+RCT*SIN(ANI)	GEO	620
63		XT2=XT+RCT*SIN(ANE)	GEO	630
64		RT2=RT+RCT*(1.0-COS(ANE))	GEO	640
65		RE=RT2+(XE-XT2)*TAN(ANE)	GEO	650
66		LT=1	GEO	660
67		DX=(XE-XI)/(LMAX-1)	GEO	670
68		IF (IUI,EQ,1) PRINT 270, XI,RI,RT,XE,RCI,RCT,ANGI,ANGE,XT,RE	GEO	680
69		IF (IUI,EQ,2) PRINT 280, XI,RI,RT,XE,RCI,RCT,ANGI,ANGE,XT,RE	GEO	690
70		DO 110 L=1,LMAX	GEO	700
71		X=XI+(L-1)*DX	GEO	710
72		IF (X,GE,XI,AND,X,LE,XTAN) GO TO 50	GEO	720
73		IF (X,GT,XTAN,AND,X,LE,XT1) GO TO 60	GEO	730
74		IF (X,GT,XT1,AND,X,LE,XT) GO TO 70	GEO	740
75		IF (X,GT,XT,AND,X,LE,XT2) GO TO 80	GEO	750
76		IF (X,GT,XT2,AND,X,LE,XE) GO TO 90	GEO	760
77	C		GEO	770
78	50	YW(L)=RI+RC*(COS(ASIN((X-XI)/RCI))-1.0)	GEO	780
79		NXNY(L)=(XI-XI)/(YW(L)-RI+RCI)	GEO	790
80		GO TO 100	GEO	800
81	C		GEO	810
82	60	YW(L)=RT1+(XT1-X)*TAN(ANI)	GEO	820
83		NXNY(L)=TAN(ANI)	GEO	830
84		GO TO 100	GEO	840
85	C		GEO	850

B.3 Inlet Boundary Conditions (inlet.f)

OCR note: this listing received a conservative normalization pass for obvious token errors; unresolved ambiguities are tracked in conversion notes.

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1  SUBROUTINE INLET
2  C *****
3  C
4  c  THIS SUBROUTINE CALCULATES THE BOUNDARY MESH POINTS AT THE NOZZLE
5  C  INLET FOR SUBSONIC FLOW
6  C
7  C *****
8  C
9  COMMON /AV/ IAV,CAV,NST,SMP,LSS,CTA,XMU,XLA,RKMU,QUT(81,21),QVT(81
10 1,21),QPT(81,21)
11 COMMON /ONESID/ UD(4),VD(4),PD(4),ROD(4)
12 COMMON /SOLUTN/ U(81,21,2),V(81,21,2),P(81,21,2),RO(81,21,2)
13 COMMON /CNTRL/ LMAX,MMAX,NMAX,NPRINT,TCONV,FDT,GAMMA,RGAS,GAMI,GA
14 1M2,L1,L2,L3,M1,M2,DX,DY,DT,N,N1,N3,NASM,IVEL,ICHAR,N1D,LJET,JFLAG,
15 2IERR,IUI,IUO,DXR,DYR,LD,MD,LMD1,LMD3,IB,RSTAR,RSTARS,NPLOT,G,PC,TC
16 3,LC,FLOW,ROLOW
17 COMMON /GEMIRYC/ NGEOM,XI,RI,XT,RT,XE,RE,RCI,RCT,ANGI,ANGE,XW(81),
18 YW(81),XWI(81),YWI(81),NXNY(81),NWPTS,IINT,IDIF,LT,NDIM
19 COMMON /GCB/ NGCB,XICB,RICB,XTCB,RTCB,XECB,RECB,RCICB,RCTCB,ANGICB
20 1,ANGECB,XCB(81),YCB(81),XCB1(81),YCB1(81),NXNYCB(81),NCBPTS,IINTCB
21 2,IDIFCB,LECB
22 COMMON /BOC/ PT(21),TT(21),THETA(21),PE,MASSE,MASSI,MASST,THRUST,N
23 1STAG
24 REAL MN3,NXNY,MASSI,MASST,NXNYCB,MASSE
25 C
26 GRGB=GAMMA*RGAS*G
27 X3=XI
28 ATERM2=0.0
29 ATERM3=0.0

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```

30 DO 180 ICHAR=1,2
31 DO 180 M=1,MMAX
32 LMN1=1+LD*(M-1)+LMD1
33 LMN3=1+LD*(M-1)+LMD3
34 LIMN1=2+LD*(M-1)+LMD1
35 LIM1N1=2+LD*(M-2)+LMD1
36 LM1N1=1+LD*(M-2)+LMD1
37 LM1N3=1+LD*M+LMD3
38 CALL MAP (2,1,M,AL,BE,DE,2,AL1,BE1,DE1)
39 U2=U(LMN1)
40 A2=SQRT(GAMMA*P(LMN1)/RO(LMN1))
41 IF (ICAR,EQ,2) GO TO 10
42 U(LMN3)=U2
43 V(LMN3)=V(LMN1)
44 A3=A2
45 C
46 C CALCULATE THE PROPERTY INTERPOLATING POLYNOMIAL COEFFICIENTS
47 C
48 10 BU=(U(LIMN1)-U(LMN1))*DXR
49 BV=(V(LIMN1)-V(LMN1))*DXR
50 BP=(P(LIMN1)-P(LMN1))*DXR
51 BRO=(RO(LIMN1)-RO(LMN1))*DXR
52 BYCB=(YCB(2)-YCB(1))*DXR
53 BAL=(AL1-AL)*DXR
54 BBE=(BE1-BE)*DXR
55 CU=U(1,M,N1)-BU*X3
56 CV=V(1,M,N1)-BV*X3
57 CP=P(1,M,N1)-BP*X3
58 CRO=RO(1,M,N1)-BRO*X3
59 CYCB=YCB(1)-BYCB*X3
60 CAL=AL-BAL*X3
61 CBE=BE-BBE*X3
62 C
63 C CALCULATE THE CROSS DERIVATIVE INTERPOLATING POLYNOMIAL
64 C COEFFICIENTS
65 C
66 IF (M,EQ,1) GO TO 20
67 DU=(U(LIMN1)-U(LIM1N1))*DYR
68 DV=(V(LIMN1)-V(LIM1N1))*DYR
69 DP=(P(LIMN1)-P(LIM1N1))*DYR
70 DRO=(RO(LIMN1)-RO(LIM1N1))*DYR
71 DU1=(U(LMN1)-U(LM1N1))*DYR
72 DV1=(V(LMN1)-V(LM1N1))*DYR
73 DP1=(P(LMN1)-P(LM1N1))*DYR
74 DRO1=(RO(LMN1)-RO(LM1N1))*DYR
75 GO TO 40
76 20 IF (NGCB,NE,0) GO TO 30
77 DU=0.0
78 DV=V(2,2,N1)*DYR
79 DP=0.0
80 DRO=0.0
81 DU1=0.0
82 DV1=V(1,2,N1)*DYR
83 DP1=0.0
84 DRO1=0.0
85 GO TO 40
86 30 DU=(U(2,2,N1)-U(2,1,N1))*DYR
87 DV=(V(2,2,N1)-V(2,1,N1))*DYR
88 DP=(P(2,2,N1)-P(2,1,N1))*DYR
89 DRO=(RO(2,2,N1)-RO(2,1,N1))*DYR
90 DU1=(U(1,2,N1)-U(1,1,N1))*DYR
91 DV1=(V(1,2,N1)-V(1,1,N1))*DYR
92 DP1=(P(1,2,N1)-P(1,1,N1))*DYR
93 DRO1=(RO(1,2,N1)-RO(1,1,N1))*DYR
94 40 BDU=(DU-DU1)*DXR
95 BDV=(DV-DV1)*DXR
96 BDP=(DP-DP1)*DXR
97 BDRO=(DRO-DRO1)*DXR
98 CDU=DU1-BDU*X3
99 CDV=DV1-BDV*X3
100 CDP=DP1-BDP*X3
101 CDRO=DRO1-BDRO*X3
102 C
103 C CALCULATE X2
104 C
105 IF (ICAR,EQ,2) A3=SQRT(GAMMA*P(LMN3)/RO(LMN3))

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```

106 DO 50 IL=1,2
107 X2=X3-(U(1,M,N3)-A3+U2-A2)*0.5*DT
108 C
109 C INTERPOLATE FOR THE PROPERTIES
110 C
111 U2=BU*X2+CU
112 P2=BP*X2+CP
113 RO2=BRO*X2+CRO
114 A2=SQRT(GAMMA*P2/RO2)
115 50 CONTINUE
116 V2=BV*X2+CV
117 YCB2=BYCB*X2+CYCB
118 AL2=BAL*X2+CAL
119 BE2=BBE*X2+CBE
120 UV2=U2*AL2+V2*BE2
121 C
122 C INTERPOLATE FOR THE CROSS DERIVATIVES
123 C
124 DU2=BDU*X2+CDU
125 DV2=BDV*X2+CDV
126 DP2=BDP*X2+CDP
127 DRO2=BDRO*X2+CDRO
128 C
129 C CALCULATE THE PSI TERMS
130 C
131 IF (NDIM,EQ,0) GO TO 70
132 IF (M,EQ,1,AND,NGCB,EQ,0) GO TO 60
133 ATERM2=RO2*V2/( (DY*(M-1))/BE2+YCB2)
134 GO TO 70
135 60 ATERM2=RO2*BE2+DV2
136 70 PSI12=-UV2+DRO2-RO2*AL2+DU2-RO2*BE2+DV2-ATERM2
137 PSI22=-UV2+DU2-AL2+DP2/RO2
138 PSI42=-UV2+DP2+A2*A2+UV2*DRO2
139 IF (ICHR,EQ,1) GO TO 130
140 C
141 C CALCULATE THE CROSS DERIVATIVES AT THE SOLUTION POINT
142 C
143 IF (M,EQ,1,AND,NGCB,EQ,0) GO TO 80
144 IF (M,EQ,MMAX) GO TO 90
145 DU3=(U(LMN3)-U(LMN3))*DYR
146 DV3=(V(LMN3)-V(LMN3))*DYR
147 DP3=(P(LMN3)-P(LMN3))*DYR
148 DRO3=(RO(LMN3)-RO(LMN3))*DYR
149 GO TO 100
150 80 DU3=0.0
151 DV3=V(1,2,N3)*DYR
152 DP3=0.0
153 DRO3=0.0
154 GO TO 100
155 90 DU3=(U(1,MMAX,N3)-U(1,M1,N3))*DYR
156 DV3=(V(1,MMAX,N3)-V(1,M1,N3))*DYR
157 DP3=(P(1,MMAX,N3)-P(1,M1,N3))*DYR
158 DRO3=(RO(1,MMAX,N3)-RO(1,M1,N3))*DYR
159 C
160 C CALCULATE THE PSI TERMS AT THE SOLUTION POINT
161 C
162 100 IF (NDIM,EQ,0) GO TO 120
163 IF (M,EQ,1,AND,NGCB,EQ,0) GO TO 110
164 ATERM3=RO(LMN3)*V(LMN3)/( (DY*(M-1))/BE+YCB(1))
165 GO TO 120
166 110 ATERM3=RO(LMN3)*BE+DV3
167 120 UV3=U(LMN3)*AL+V(LMN3)*BE
168 PSI13=-UV3+DRO3-RO(LMN3)*AL+DU3-RO(LMN3)*BE+DV3-ATERM3
169 PSI23=-UV3+DU3-AL*DP3/RO(LMN3)
170 PSI43=-UV3+DP3+A3*A3+UV3*DRO3
171 GO TO 140
172 130 PSI23=PSI22
173 PSI43=PSI42
174 PSI13=PSI12
175 C
176 C SOLVE THE COMPATIBILITY EQUATIONS FOR U,V,P, AND RO
177 C
178 140 MN3=SQRT(U(LMN3)*U(LMN3)+V(LMN3)*V(LMN3))/A3
179 T2=P2/(RO2*RGAS*G)
180 PSI1B=(PSI12+PSI13)*0.5
181 PSI2B=(PSI22+PSI23)*0.5

```

```

182 PSI4B=(PSI42+PSI43)*0.5
183 GPSI1B=GAMMA*PSI1B
184 TTTHETA=TAN(THETA(M))
185 UCORR=0.5+0.5/SQRT(1.0+TTTHETA*TTTHETA)
186 C
187 DO 160 ITER=1,20
188 DEM=(1.0+GAM2*MN3*MN3)
189 P(LMN3)=PT(M)/(DEM*GAM1)
190 T3=TT(M)/DEM
191 PB=(P2+P(LMN3))*0.5
192 RTB=RGAS*(T2+T3)*0.5*G
193 U(LMN3)=U2+DT*PSI2B+(P(LMN3)-P2-(PSI4B+RTB*GPSI1B)*DT)*SQRT(RTB/GA
194 MMA)/PB
195 U(LMN3)=U(LMN3)*UCORR
196 V(LMN3)=-U(LMN3)*TTTHETA
197 OMN3=MN3
198 MN3=SQRT((U(LMN3)*U(LMN3)+V(LMN3)*V(LMN3))/(T3*GRGB))
199 IF (OMN3,NE,0.0) GO TO 150
200 IF (ABS(MN3-OMN3),LE,0.0001) GO TO 170
201 GO TO 160
202 150 IF (ABS((MN3-OMN3)/OMN3),LE,0.001) GO TO 170
203 160 CONTINUE
204 C
205 PRINT 190, M,N
206 170 RO(LMN3)=P(LMN3)/(RGAS*T3*G)
207 180 CONTINUE
208 RETURN
209 C
210 190 FORMAT (1H0,58H***** THE SOLUTION FOR NOZZLE ENTRANCE BOUNDARY POI
211 1NT ( 1,,I2,1H,,I4,43H) FAILED TO CONVERGE IN 20 ITERATIONS *****
212 END

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B.4 Wall Boundary Conditions (wall.f)

OCR note: this listing received a conservative normalization pass for obvious token errors; unresolved ambiguities are tracked in conversion notes.

```

1 SUBROUTINE WALL
2 C
3 C *****
4 C
5 C THIS SUBROUTINE CALCULATES THE BOUNDARY MESH POINTS AT THE NOZZLE
6 C WALL, EXHAUST JET BOUNDARY, AND CENTERBODY
7 C
8 C *****
9 C
10 COMMON /AV/ IAV,CAV,NST,SMP,LSS,CTA,XMU,XLA,RKMU,QUT(81,21),QVT(81
11 1,21),QPT(81,21)
12 COMMON /ONESID/ UD(4),VD(4),PD(4),ROD(4)
13 COMMON /SOLUTN/ U(81,21,2),V(81,21,2),P(81,21,2),RO(81,21,2)
14 COMMON /CNTRL/ LMAX,MMAX,NMAX,NPRINT,TCONV,FDT,GAMMARGAS,GAM1,GA
15 1M2,L1,L2,L3,M1,M2,DX,DY,DT,N,N1,N3,NASM,IVEL,ICHAR,NID,LJET,JFLAG,
16 2IERR,IUI,IUO,DXR,DYR,LD,MD,LMD1,LMD3,IB,RSTAR,RSTARS,NPLOT,G,PC,TC
17 3,LC,FLOW,ROLOW
18 COMMON /GEMTRYC/ NGEOM,XI,RI,XT,RT,XE,RE,RCI,RCT,ANGL,ANGE,XW(81),
19 1YW(81),XWI(81),YWI(81),NXNY(81),NWPTS,IINT,IDIF,LT,NDIM
20 COMMON /GCB/ NGCB,XICB,RICB,XTCB,RTCB,XECB,RECB,RCICB,RCTCB,ANGICB
21 2,ANGECEB,XCB(81),YCB(81),XCB1(81),YCB1(81),NXNYCB(81),NCBPTS,IINTCB
22 3,IDIFCB,LECB
23 COMMON /BCC/ PT(21),TT(21),THETA(21),PE,MASSE,MASSI,MASST,THRUST,N
24 1STAG
25 REAL MN3,NXNY,MASSI,MASST,NXNYCB,MASSE
26 C
27 IF (N.EQ.1) DELY=0.005
28 XWID=0.0
29 IF (IB.EQ.1) GO TO 10
30 Y1=0.0 $ Y3=0.0 $ MDUM1=1 $ MDUM1=2 $ SIGN=-1.0
31 GO TO 20
32 10 Y1=1.0 $ Y3=1.0 $ MDUM1=MAX $ MDUM1=M1 $ SIGN=1.0
33 20 ATERM2=0.0
34 ATERM3=0.0

```

```

35 LDUM=LMAX
36 IF (ICAR.EQ.2) LDUM=L1
37 LMDM=LD*(MDUM-1)
38 LMDM1=LD*(MDUM-1)
39 DYS=SIGN*DYS
40 DO 350 L=2,LDUM
41 LMN1=L+LMDM+LMD1
42 LMN3=L+LMDM+LMD3
43 LMN11=L+LMDM1+LMD1
44 L1MN1=L-1+LMDM+LMD1
45 L1MN3=L+1+LMDM+LMD3
46 L1MN11=L-1+LMDM1+LMD1
47 IF (JFLAG.EQ.0) GO TO 50
48 IF (IB.EQ.2) GO TO 50
49 C
50 XWID=XWI(L)
51 IF (ICAR.EQ.1) GO TO 30
52 C
53 C USE THE DUMMY ARRAYS TO MANIPULATE THE ONE-SIDED SOLUTIONS
54 C
55 IF (L.NE.LJET-2) GO TO 30
56 U(LMN3)=UD(3)
57 V(LMN3)=VD(3)
58 P(LMN3)=PD(3)
59 RO(LMN3)=ROD(3)
60 GO TO 50
61 30 IF (L.NE.LJET-1) GO TO 40
62 IF (ICAR.EQ.1) UOLD=U(LMN1)
63 U(LMN1)=UD(1)
64 V(LMN1)=VD(1)
65 P(LMN1)=PD(1)
66 RO(LMN1)=ROD(1)
67 GO TO 50
68 40 IF (L.EQ.LJET) GO TO 50
69 U(LMN1)=UD(2)
70 V(LMN1)=VD(2)
71 P(LMN1)=PD(2)
72 RO(LMN1)=ROD(2)
73 C
74 50 U1=U(LMN1)
75 V1=V(LMN1)
76 P1=P(LMN1)
77 RO1=RO(LMN1)
78 U2=U1
79 V2=V1
80 A1=SQRT(GAMMA*P1/RO1)
81 A2=A1
82 IF (ICAR.EQ.2) GO TO 60
83 U3=U1
84 V3=V1
85 P3=P1
86 RO3=RO1
87 A3=A1
88 GO TO 70
89 60 U3=U(LMN3)
90 V3=V(LMN3)
91 P3=P(LMN3)
92 RO3=RO(LMN3)
93 A3=SQRT(GAMMA*P3/RO3)
94 C
95 C CALCULATE THE PROPERTY INTERPOLATING POLYNOMIAL COEFFICIENTS
96 C
97 70 BU=(U1-U(LMN1))*DYS
98 BV=(V1-V(LMN1))*DYS
99 BP=(P1-P(LMN1))*DYS
100 BRO=(RO1-RO(LMN1))*DYS
101 CU=U1-BU*Y3
102 CV=V1-BV*Y3
103 CP=P1-BP*Y3
104 CRO=RO1-BRO*Y3
105 C
106 C CALCULATE THE CROSS DERIVATIVE INTERPOLATING POLYNOMIAL
107 C COEFFICIENTS
108 C
109 DU=(U1-U(LMN1))*DXR
110 DV=(V1-V(LMN1))*DXR

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111 DP= (P1-P(L1MIN1)) *DXR
112 DRO= (RO1-RO(L1MIN1)) *DXR
113 DU1= (U(L1MIN1)-U(L1MIN1)) *DXR
114 DV1= (V(L1MIN1)-V(L1MIN1)) *DXR
115 DP1= (P(L1MIN1)-P(L1MIN1)) *DXR
116 DRO1= (RO(L1MIN1)-RO(L1MIN1)) *DXR
117 BDU= (DU-DU1) *DYS
118 BDV= (DV-DV1) *DYS
119 BDP= (DP-DP1) *DYS
120 BDRO= (DRO-DRO1) *DYS
121 CDU= DU-BDU*Y3
122 CDV= DV-BDV*Y3
123 CDP= DP-BDP*Y3
124 CDRO= DRO-BDRO*Y3
125 C
126 C CALCULATE Y2
127 C
128 CALL MAP (1,L,MDUM,AL,BE,DE,LD1,AL1,BE1,DE1)
129 ALS=SQRT(AL*AL+BE*BE)
130 UV3=U3*AL+V3*BE+DE
131 AL2=AL
132 DO 90 ILL=1,3
133 UV2=U2*AL2+V2*BE+DE
134 Y2=Y3-(UV2+SIGN*AL*ALS*A2+UV3+SIGN*ALS*A3)*DT*0.5
135 C
136 C INTERPOLATE FOR THE PROPERTIES
137 C
138 U2=BU*Y2+CU
139 V2=BV*Y2+CV
140 P2=BP*Y2+CP
141 RO2=BRO*Y2+CRO
142 AL2=Y2*AL
143 AD=GAMMA*P2/RO2
144 IF (AD.GT.0.0) GO TO 80
145 PRINT 360,N,L,MDUM
146 IERR=1
147 RETURN
148 80 A2=SQRT(AD)
149 90 CONTINUE
150 C
151 C INTERPOLATE FOR THE CROSS DERIVATIVES
152 C
153 DU1=DU
154 DV1=DV
155 DP1=DP
156 DRO1=DRO
157 DU2=BDU*Y2+CDU
158 DV2=BDV*Y2+CDV
159 DP2=BDP*Y2+CDP
160 DRO2=BDRO*Y2+CDRO
161 C
162 C CALCULATE THE PSI TERMS
163 C
164 IF (NDIM.EQ.0) GO TO 110
165 IF (IB.EQ.2) GO TO 100
166 ATERM2=RO2*V2/(YCB(L)+Y2/BE)
167 GO TO 110
168 100 ATERM2=RO2*V2/(YCB(L)+Y2/BE)
169 IF (IAV.EQ.0) GO TO 110
170 ATDS=RO2*V(L,2,N1)*DYS*BE
171 IF (ABS(ATERM2).GT.ABS(ATDS)) ATERM2=ATDS
172 C
173 110 PSI21=-U1*DU1-DP1/RO1
174 PSI31=-U1*DV1
175 PSI41=-U1*DP1+A1*A1*U1*DRO1
176 PSI12=-U2*DRO2-RO2*DU2-ATERM2
177 PSI22=-U2*DU2-DP2/RO2
178 PSI32=-U2*DV2
179 PSI42=-U2*DP2+A2*A2*U2*DRO2
180 IF (ICAR.EQ.1) GO TO 150
181 C
182 C CALCULATE THE CROSS DERIVATIVES AT THE SOLUTION POINT
183 C
184 IF (JFLAG.EQ.0) GO TO 120
185 IF (IB.EQ.2) GO TO 120
186 IF (L.EQ.2) GO TO 120

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187 IF (L.NE.LJET-1) GO TO 120
188 IF (LJET.EQ.2) GO TO 120
189 GO TO 130
190 120 DU3=(U(LIMN3)-U3)*DXR
191 DV3=(V(LIMN3)-V3)*DXR
192 DP3=(P(LIMN3)-P3)*DXR
193 DRO3=(RO(LIMN3)-RO3)*DXR
194 GO TO 140
195 130 DU3=(U3-U(L-1,MDUMN3))*DXR
196 DV3=(V3-V(L-1,MDUMN3))*DXR
197 DP3=(P3-P(L-1,MDUMN3))*DXR
198 DRO3=(RO3-RO(L-1,MDUMN3))*DXR
199 C
200 C ENTER THE EXHAUST JET ITERATION LOOP
201 C
202 140 IF (JFLAG.EQ.0) GO TO 150
203 IF (IB.EQ.2) GO TO 150
204 IF (L.LT.LJET) GO TO 150
205 YWI(L)=YW(L)
206 UDUM=U(LMN3)
207 VDUM=V(LMN3)
208 PDUM=P(LMN3)
209 RODUM=RO(LMN3)
210 150 DO 290 NJ=1,10
211 IF (ICHR.EQ.1) GO TO 250
212 IF (JFLAG.EQ.0) GO TO 210
213 IF (IB.EQ.2) GO TO 210
214 IF (L.LT.LJET) GO TO 210
215 IF (NJ.EQ.1) GO TO 200
216 IF (NJ.GT.2) GO TO 180
217 160 YWOLD=YW(L)
218 POLD=P(LMN3)
219 IF (P(LMN3).LT.PE) GO TO 170
220 YW(L)=YW(L)+DELY
221 GO TO 190
222 170 YW(L)=YW(L)-DELY
223 GO TO 190
224 180 IF (P(LMN3).EQ.POLD) GO TO 160
225 DYDP=(YW(L)-YWOLD)/(P(LMN3)-POLD)
226 YWNEW=YW(L)+DYDP*(PE-P(LMN3))
227 YWOLD=YW(L)
228 POLD=P(LMN3)
229 YW(L)=YWNEW
230 190 IF (YW(L).LT.(0.98*YWOLD)) YW(L)=0.98*YWOLD
231 IF (YW(L).GT.(1.02*YWOLD)) YW(L)=1.02*YWOLD
232 200 NXNY(L)=- (YW(L)-YW(L-1))*DXR
233 XWI(L)=(YW(L)-YWI(L))/DT
234 XWID=XWI(L)
235 CALL MAP (1,L,MDUM,AL,BE,DE,LD1,AL1,BE1,DE1)
236 ALS=SQRT(AL*AL+BE*BE)
237 U(LMN3)=UDUM
238 V(LMN3)=VDUM
239 P(LMN3)=PDUM
240 RO(LMN3)=RODUM
241 C
242 C CALCULATE THE PSI TERMS AT THE SOLUTION POINT
243 C
244 210 IF (NDIM.EQ.0) GO TO 240
245 IF (IB.EQ.2) GO TO 220
246 ATERM3=RO3*V2/(YCB(L)+1.0/BE)
247 GO TO 240
248 220 IF (YCB(L).EQ.0.0) GO TO 230
249 ATERM3=RO3*V3/YCB(L)
250 IF (IAV.EQ.0) GO TO 240
251 ATDS=RO3*V(L,2,N3)*DYS*BE
252 IF (ABS(ATERM3).GT.ABS(ATDS)) ATERM3=ATDS
253 GO TO 240
254 230 ATERM3=RO3*V(L,2,N3)*DYS*BE
255 C
256 240 PSI13=-U3*DRO3-RO3*DU3-ATERM3
257 PSI23=-U3*DU3-DP3/RO3
258 PSI33=-U3*DV3
259 PSI43=-U3*DP3+A3*A3*U3*DRO3
260 C CALCULATE THE COMPATIBILITY EQUATION COEFFICIENTS
261 C
262 250 ABR=NXNY(L)

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263 IF (IB.EQ.2) ABR=NXNYCB(L)
264 ALB=0.5*(AL2+AL)/ALS
265 BEB=BE/ALS
266 A1B=(A1+A3)*0.5
267 A2B=(A2+A3)*0.5
268 RO1B=(RO1+RO3)*0.5
269 RO2B=(RO2+RO3)*0.5
270 IF (ICHR.EQ.1) GO TO 260
271 PSI21B=(PSI21+PSI23)*0.5
272 PSI31B=(PSI31+PSI33)*0.5
273 PSI41B=(PSI41+PSI43)*0.5
274 PSI12B=(PSI12+PSI13)*0.5
275 PSI22B=(PSI22+PSI23)*0.5
276 PSI32B=(PSI32+PSI33)*0.5
277 PSI42B=(PSI42+PSI43)*0.5
278 GO TO 270
279 260 PSI21B=PSI21
280 PSI31B=PSI31
281 PSI41B=PSI41
282 PSI12B=PSI12
283 PSI12B=PSI12
284 PSI22B=PSI22
285 PSI32B=PSI32
286 PSI42B=PSI42
287 C
288 C SOLVE THE COMPATIBILITY EQUATIONS FOR U,V,P, AND RO
289 C
290 270 U(LMN3)=(U(LMN1)-ABR*(V(LMN1)-XWID)+(PSI21B-ABR*PSI31B)*DT)/(1.0+A
291 1BR*ABR)
292 V(LMN3)=-U(LMN3)*ABR*XWID
293 P(LMN3)=P2-SIGN*RO2B*A2B*(ALB*(U(LMN3)-U2)+BEB*(V(LMN3)-V2))+(PSI4
294 22B+A2B*A2B*PSI12B+SIGN*RO2B*A2B*(ALB*PSI22B+BEB*PSI32B))*DT
295 IF (P(LMN3).LE.0.0) P(LMN3)=FLOW*PC
296 RO(LMN3)=RO(LMN1)+(P(LMN3)-P(LMN1)-PSI41B*DT)/(A1B*A1B)
297 IF (RO(LMN3).LE.0.0) RO(LMN3)=ROLOW/G
298 IF (IAV.EQ.0) GO TO 280
299 C
300 C ADD THE ARTIFICIAL VISCOSITY FOR SHOCK CALCULATIONS
301 C
302 IF (ICHR.EQ.1) GO TO 280
303 U(LMN3)=U(LMN3)+(QUT(L,MJUM)-ABR*QVT(L,MJUM))/(1.0+ABR*ABR)
304 V(LMN3)=-U(LMN3)*ABR
305 P(LMN3)=P(LMN3)+QPT(L,MJUM)
306 280 IF (JFLAG.EQ.0) GO TO 350
307 IF (IB.EQ.2) GO TO 350
308 IF (L.LT.LJET-1) GO TO 350
309 IF (L.EQ.LJET-1) GO TO 300
310 IF (ICHR.EQ.1) GO TO 350
311 DELP=ABS((P(LMN3)-PE)/PE)
312 IF (DELP.LE.0.001) GO TO 350
313 290 CONTINUE
314 GO TO 350
315 C
316 C SOLVE THE COMPATIBILITY EQUATIONS FOR THE DOWNSTREAM SIDE OF THE
317 C NOZZLE WALL EXIT POINT
318 C
319 300 UD(3)=U(LMN3)
320 VD(3)=V(LMN3)
321 PD(3)=P(LMN3)
322 ROD(3)=RO(LMN3)
323 PD(4)=PE
324 XMI=SQRT((UD(3)*UD(3)+VD(3)*VD(3))/(GAMMA*PD(3)/ROD(3)))
325 DUMD=1.0+GAM2*XMI*XMI
326 TD=PD(3)/ROD(3)/RGAS/G
327 TTD=TD*DUMD
328 IF (PE.GT.PD(3).AND.XMI.GE.1.0) GO TO 310
329 TTD=TD*DUMD
330 IF (PE.GT.PD(3).AND.XMI.GE.1.0) GO TO 310
331 PTD=PD(3)*DUMD*GAMI
332 ROD(4)=ROD(3)*(PE/PD(3))**(1.0/GAMMA)
333 GO TO 320
334 310 PRD=PE/PD(3)
335 GAMD=(GAMMA+1.0)/(GAMMA-1.0)
336 ROD(4)=ROD(3)*(GAMD*PRD+1.0)/(PRD+GAMD)
337 320 TE=PE/ROD(4)/RGAS/G
338 XMACH=SQRT((TTD/TE-1.0)/GAM2)

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339      SS=SQRT(GAMMA*PE/ROD(4))
340      VMAG=XMACH*SS
341      UD(4)=VMAG/SQRT(1.0+NXNY(LJET)*NXNY(LJET))
342      VD(4)=-UD(4)*NXNY(LJET)
343 C
344 C   AVERAGE THE 1-SIDED MACH NOS FOR THE INTERIOR POINT CALCULATIONS
345 C
346      XM2=SQRT((UD(4)*UD(4)+VD(4)*VD(4))/(GAMMA*PD(4)/ROD(4)))
347      IF (XM1.GE.1.0) GO TO 350
348      XMB=(XM1+XM2)/2.0
349      IF (XMB.GE.1.0) GO TO 330
350      DPL=1.0
351      DPR=1.0
352      GO TO 340
353 330 DPL=XM2-1.0
354      DPR=1.0-XM1
355      XMB=1.0
356 340 DPLR=DPR+DPL
357      DUM=1.0+GAM2*XMB*XMB
358      TEMP=TTD/DUM
359      P(LMN3)=PTD/DUM**GAM1
360      RO(LMN3)=P(LMN3)/(RGAS*TEMP*G)
361      QA=SQRT(2.0*GAM1*(RGAS*TTD*G-P(LMN3)/RO(LMN3)))
362      DNXY=(DPR*NXNY(LJET)+DPL*NXNY(L))/DPLR
363      U(LMN3)=QA/SQRT(1.0+DNXY*DNXY)
364      V(LMN3)=-U(LMN3)*DNXY
365      IF (ICHAR.EQ.1) GO TO 350
366      UD(1)=UD(3)
367      VD(1)=VD(3)
368      PD(1)=PD(3)
369      ROD(1)=ROD(3)
370      UD(2)=UD(4)
371      VD(2)=VD(4)
372      PD(2)=PD(4)
373      ROD(2)=ROD(4)
374 350 CONTINUE
375      IF (JFLAG.EQ.0) RETURN
376      IF (IB.EQ.2) RETURN
377      IF (ICHAR.EQ.1) RETURN
378      U(LJET-1,MMAX,N1)=UOLD
379      YWI(LMAX)=YW(LMAX)
380      YW(LMAX)=2.0*YW(L1)-YW(L2)
381      NXNY(LMAX)=- (YW(LMAX)-YW(L1))*DXR
382      XWI(LMAX)=(YW(LMAX)-YWI(LMAX))/DT
383      DELY=ABS(YW(LJET)-YWI(LJET))
384      IF (DELY.EQ.0.0) DELY=0.0001
385      RETURN
386 C
387 360 FORMAT (1H0,61H***** A NEGATIVE SQUARE ROOT OCCURED IN SUBROUTINE
388      1WALL AT N=,I4,4H, L=,I2,8H, AND M=,I2,6H ***** )
389      END

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B.5 Interior Mesh Calculations (inter.f)

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1      SUBROUTINE INTER
2 C
3 C   *****
4 C
5 C   THIS SUBROUTINE CALCULATES THE NOZZLE RADIUS AND OUTER NORMAL
6 C
7 C   *****
8 C
9      COMMON /AV/ IAV,CAV,NST,SMP,LSS,CTA,XMU,XLA,RKMU,QUT(81,21),QVT(81
10      1,21),QPT(81,21)
11      COMMON /ONESID/ UD(4),VD(4),PD(4),ROD(4)
12      COMMON /SOLUTN/ U(81,21,2),V(81,21,2),P(81,21,2),RO(81,21,2)
13      COMMON /CNTRLC/ LMAX,MMAX,NMAX,NPRINT,TCNV,FDT,GAMMA,RGAS,GAM1,GA
14      1M2,L1,L2,L3,M1,M2,DX,DY,DT,N,N1,N3,NASM,IVEL,ICHAR,NID,LJET,JFLAG,
15      2IERR,IUI,IUO,DXR,DYR,LD,MD,LMD1,LMD3,IB,RSTAR,RSTARS,NPLOT,G,PC,TC
16      3,LC,FLOW,ROLOW
17      COMMON /GEMTRYC/ NGEOM,XI,RI,XT,RT,XE,RE,RCI,RCT,ANGI,ANGE,XW(81),
18      1YW(81),XWI(81),YWI(81),NXNY(81),NWPTS,IINT,IDIF,LT,NDIM

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19      COMMON /GCB/ NGCB,XICB,RICB,XTCB,RTCB,XECB,RECB,RCICB,RCTCB,ANGICB
20      2,ANGECEB,XCB(81),YCB(81),XCBI(81),YCBI(81),NXNYCB(81),NCBPTS,IINTCB
21      3,IDIFCB,LECB
22      COMMON /BCC/ PT(21),TT(21),THETA(21),PE,MASSE,MASSI,MASST,THRUST,N
23      1STAG
24      REAL MN3,NXNY,MASSI,MASST,NXNYCB,MASSE
25 C
26      ATERM=0.0
27      IF (ICBAR,EQ,2) GO TO 40
28 C
29 C      COMPUTE THE TENTATIVE SOLUTION AT T+DT
30 C
31      MDUM=1
32      IF (NGCB.NE.0) MDUM=2
33      DO 30 L=2,LMAX
34      DO 30 M=MDUM,M1
35      CALL MAP (1,L,M,AL,BE,DE,LD1,AL1,BE1,DE1)
36      LMD2=LD*(M-1)
37      LMN1=L+LMD2+LMD1
38      LMN3=L+LMD2+LMD3
39      L1MN1=L-1+LMD2+LMD1
40      LMN1=L+LD*(M-2)+LMD1
41      UB=U(LMN1)
42      VB=V(LMN1)
43      PB=P(LMN1)
44      ROB=RO(LMN1)
45      ASB=GAMMA*PB/ROB
46      IF (M.NE.1) GO TO 10
47      DUDX=(UB-U(L1MN1))*DXR
48      DPDX=(PB-P(L1MN1))*DXR
49      DRODX=(ROB-RO(L1MN1))*DXR
50      DVDY=(4.0*V(L,2,N1)-V(L,3,N1))*0.5*DYR
51      V(LMN3)=0.0
52 C
53      URHS=-UB*DUDX-DPDX/ROB
54      RORHS=-UB*DRODX-ROB*DUDX-(1+NDIM)*ROB*BE+DVDY
55      PRHS=-UB*DPDX+ASB*(RORHS+UB*DRODX)
56      GO TO 20
57 10 IF (NDIM.EQ.1) ATERM=ROB*VB/((M-1)*DY/BE+YCB(L))
58      UVB=UB+AL*VB*BE+DE
59      DUDX=(UB-U(L1MN1))*DXR
60      DVDX=(VB-V(L1MN1))*DXR
61      DPDX=(PB-P(L1MN1))*DXR
62      DRODX=(ROB-RO(L1MN1))*DXR
63      DUDY=(UB-U(LMN1))*DYR
64      DVDY=(VB-V(LMN1))*DYR
65      DPDY=(PB-P(LMN1))*DYR
66      DRODY=(ROB-RO(LMN1))*DYR
67 C
68      URHS=-UB*DUDX-UVB*DUDY-(DPDX+AL*DPDY)/ROB
69      VRHS=-UB*DVDX-UVB*DVDY-BE*DPDY/ROB
70      RORHS=-UB*DRODX-UVB*DRODY-ROB*(DUDX+AL*DUDY+BE+DVDY)-ATERM
71      PRHS=-UB*DPDX-UVB*DPDY+ASB*(RORHS+UB*DRODX+UVB*DRODY)
72      V(LMN3)=V(LMN1)+VRHS*DT
73 20 U(LMN3)=U(LMN1)+URHS*DT
74      P(LMN3)=P(LMN1)+PRHS*DT
75      RO(LMN3)=RO(LMN1)+RORHS*DT
76      IF (P(LMN3).LE.0.0) P(LMN3)=FLOW*PC
77      IF (RO(LMN3).LE.0.0) RO(LMN3)=ROLOW/G
78 30 CONTINUE
79      RETURN
80 C
81 C      COMPUTE THE FINAL SOLUTION AT T+DT
82 C
83 40 MDUM=1
84      IF (NGCB.NE.0) MDUM=2
85      DO 70 L=2,L1
86      DO 70 M=MDUM,M1
87      CALL MAP (1,L,M,AL,BE,DE,LD1,AL1,BE1,DE1)
88      LMD2=LD*(M-1)
89      LMN1=L+LMD2+LMD1
90      LMN3=L+LMD2+LMD3
91      L1MN3=L+1+LMD2+LMD3
92      LMN3=L+LD*M+LMD3
93      UB=U(LMN3)
94      VB=V(LMN3)

```

```

95 PB=P(LMN3)
96 ROB=RO(LMN3)
97 ASB=GAMMA*PB/ROB
98 IF (M.NE.1) GO TO 50
99 DUDX=(U(L1MN3)-UB)*DXR
100 DPDX=(P(L1MN1)-PB)*DXR
101 DRDXX=(RO(L1MN1)-ROB)*DXR
102 DVDY=(4.0*V(L,2,N3)-V(L,3,N3))*0.5*DYR
103 V(LMN3)=0.0
104 C
105 URHS=-UB*DUDX-DPDX/ROB
106 RORHS=-UB*DRDXX-ROB*DUDX*(1+NDIM)*ROB*BE*DVDY
107 PRHS=-UB*DPDX+ASB*(RORHS+UB*DRDXX)
108 GO TO 60
109 50 IF (NDIM.EQ.1) ATERM=ROB*VB/((M-1)*DY/BE+YCB(L))
110 UVB=UB+AL*VB*BE+DE
111 DUDX=(U(L1MN3)-UB)*DXR
112 DVDX=(V(L1MN3)-VB)*DXR
113 DPDX=(P(L1MN3)-PB)*DXR
114 DRDXX=(RO(L1MN3)-ROB)*DXR
115 DUDY=(U(L1MN3)-UB)*DYR
116 DVDY=(V(L1MN3)-VB)*DYR
117 DPDY=(P(L1MN3)-PB)*DYR
118 DRDXY=(RO(L1MN3)-ROB)*DYR
119 C
120 URHS=-UB*DUDX-UVB*DUDY-(DPDX+AL*DPDY)/ROB
121 VRHS=-UB*DVDX-UVB*DVDY-BE*DPDY/ROB
122 RORHS=-UB*DRDXX-UVB*DRDXY-ROB*(DUDX+AL*DUDY+BE*DVDY)-ATERM
123 PRHS=-UB*DPDX-UVB*DPDY+ASB*(RORHS+UB*DRDXX+UVB*DRDXY)
124 V(LMN3)=(V(LMN1)+V(LMN3)+VRHS*DT)*0.5
125 60 U(LMN3)=(U(LMN1)+U(LMN3)+URHS*DT)*0.5
126 P(LMN3)=(P(LMN1)+P(LMN3)+PRHS*DT)*0.5
127 RO(LMN3)=(RO(LMN1)+RO(LMN3)+RORHS*DT)*0.5
128 IF (P(LMN3).LE.0.0) P(LMN3)=PLOW*PC
129 IF (RO(LMN3).LE.0.0) RO(LMN3)=ROLOW/G
130 IF (IAV.EQ.0) GO TO 70
131 C
132 C ADD THE ARTIFICIAL VISCOSITY FOR SHOCK CALCULATIONS
133 C
134 U(LMN3)=U(LMN3)+QUT(L,M)
135 V(LMN3)=V(LMN3)+QVT(L,M)
136 IF (M.EQ.1) V(LMN3)=0.0
137 P(LMN3)=P(LMN3)+QPT(L,M)
138 70 CONTINUE
139 RETURN
140 END

```

B.6 Mass Flow Calculations (masflo.f)

```

1 SUBROUTINE MASFLO(ISURF) MAS 10
2 C MAS 20
3 C ***** MAS 30
4 C MAS 40
5 C THIS SUBROUTINE CALCULATES THE INITIAL-DATA OR SOLUTION SURFACE MAS 50
6 C MASS FLOW AND THRUST MAS 60
7 C MAS 70
8 C ***** MAS 80
9 C MAS 90
10 COMMON /AV/ IAV,CAV,NST,SMP,LSS,CTA,XMU,XLA,RKMU,QUT(81,21),QVT(81,MAS 100
11 1,21),QPT(81,21) MAS 110
12 COMMON /ONESID/ UD(4),VD(4),PD(4),ROD(4) MAS 120
13 COMMON /SOLUTN/ U(81,21,2),V(81,21,2),P(81,21,2),RO(81,21,2) MAS 130
14 COMMON /CNTRL/ LMAX,MMAX,NMAX,NPRINT,TCONV,FDT,GAMMA,ARGAS,GAMI,GAMAS 140
15 1M2,L1,L2,L3,M1,M2,DX,DY,DT,N,N1,N3,NASM,IVEL,ICHA,NID,LJET,JFLAG,MAS 150
16 2IERR,IUI,IUO,DXR,DYR,LD,MD,LMD1,LMD3,IB,RSTAR,RSTARS,NPLOT,G,PC,TCMAS 160
17 3,LC,PLOW,ROLOW MAS 170
18 COMMON /GEMTRYC/ NGEOM,XI,RI,XT,RT,XE,RE,RCI,RCT,ANGI,ANGE,XW(81),MAS 180
19 1YWI(81),XWI(81),YWI(81),NXNY(81),NWPTS,IINT,IDI,LT,NDIM MAS 190
20 COMMON /GCB/ NGCB,XICB,RCB,XTCB,RTCB,XECB,RECB,RCICB,RCTCB,ANGICBMAS 200
21 1,ANGE,XCBI(81),YCB(81),XCBI(81),YCB(81),NXNYCB(81),NCBPTS,IINTCBMAS 210
22 2,IDI,FCB,LECB MAS 220
23 COMMON /BOC/ PT(21),TT(21),THETA(21),PE,MASSE,MASSI,MASST,THRUST,NMAS 230

```

24	3STAG	MAS	240
25	REAL MN3,NXNY,MASSI,MASST,NXNYCB,MASSE	MAS	250
26	C	MAS	260
27	LC2=LC*LC	MAS	270
28	LDUM=LMAX-1	MAS	280
29	IF (LT,EQ,LMAX) LT=LMAX-1	MAS	290
30	IF (JFLAG,NE,0) LDUM=LJET-1	MAS	300
31	IF (ISURF,EQ,1,OR,NID,EQ,0) GO TO 30	MAS	310
32	C	MAS	320
33	C CALCULATE THE MASS FLOW AND THRUST FOR THE 1-D INITIAL-DATA	MAS	330
34	C SURFACE	MAS	340
35	C	MAS	350
36	IF (NDIM,EQ,1) GO TO 10	MAS	360
37	AREAI=(YW(1)-YCB(1))/LC2	MAS	370
38	AREAT=(YW(LT)-YCB(LT))/LC2	MAS	380
39	AREAE=(YW(LDUM)-YCB(LDUM))/LC2	MAS	390
40	GO TO 20	MAS	400
41	10 AREAI=3.141593*(YW(1)**2-YCB(1)**2)/LC2	MAS	410
42	AREAT=3.141593*(YW(LT)**2-YCB(LT)**2)/LC2	MAS	420
43	AREAE=3.141593*(YW(LDUM)**2-YCB(LDUM)**2)/LC2	MAS	430
44	GO TO 20	MAS	440
45	20 VMI=SQRT(U(1,1,1)**2+V(1,1,1)**2)	MAS	450
46	VMT=SQRT(U(LT,1,1)**2+V(LT,1,1)**2)	MAS	460
47	VME=SQRT(U(LDUM,1,1)**2+V(LDUM,1,1)**2)	MAS	470
48	MASSI=RO(1,1,1)*VMI*AREAI*G	MAS	480
49	MASST=RO(LT,1,1)*VMT*AREAT*G	MAS	490
50	MASSE=RO(LDUM,1,1)*VME*AREAE*G	MAS	500
51	THRUST=RO(LDUM,1,1)*U(LDUM,1,1)**2*AREAE	MAS	510
52	C	MAS	520
53	C CALCULATE THE MASS FLOW AND THRUST FOR THE 2-D INITIAL-DATA	MAS	530
54	C SURFACE	MAS	540
55	C	MAS	550
56	30 MASSI=0.0	MAS	560
57	MASST=0.0	MAS	570
58	MASSE=0.0	MAS	580
59	THRUST=0.0	MAS	590
60	DYI=DY*(YW(1)-YCB(1))	MAS	600
61	DYT=DY*(YW(LT)-YCB(LT))	MAS	610
62	DYE=DY*(YW(LDUM)-YCB(LDUM))	MAS	620
63	ND=1	MAS	630
64	IF (ISURF,EQ,1) ND=N3	MAS	640
65	DO 60 M=1,M1	MAS	650
66	RADI=(M-1)*DYI+YCB(1)	MAS	660
67	RADT=(M-1)*DYT+YCB(LT)	MAS	670
68	RADE=(M-1)*DYE+YCB(LDUM)	MAS	680
69	IF (NDIM,EQ,1) GO TO 40	MAS	690
70	AREAI=DYI/LC2	MAS	700
71	AREAT=DYT/LC2	MAS	710
72	AREAE=DYE/LC2	MAS	720
73	GO TO 50	MAS	730
74	40 AREAI=3.141593*((RADI+DYI)**2-RADI**2)/LC2	MAS	740
75	AREAT=3.141593*((RADT+DYT)**2-RADT**2)/LC2	MAS	750
76	AREAE=3.141593*((RADE+DYE)**2-RADE**2)/LC2	MAS	760
77	ROUI=(RO(1,M,ND)*U(1,M,ND)+RO(1,M+1,ND)*U(1,M+1,ND))*0.5	MAS	770
78	ROUT=(RO(LT,M,ND)*U(LT,M,ND)+RO(LT,M+1,ND)*U(LT,M+1,ND))*0.5	MAS	780
79	ROUE=(RO(LDUM,M,ND)*U(LDUM,M,ND)+RO(LDUM,M+1,ND)*U(LDUM,M+1,ND))*0.5	MAS	790
80	1.5	MAS	800
81	ROUE2=(RO(LDUM,M,ND)*U(LDUM,M,ND)**2+RO(LDUM,M+1,ND)*U(LDUM,M+1,ND)**2)*0.5	MAS	810
82	1)**2)*0.5	MAS	820
83	MASSI=MASSI+ROUI*AREAI*G	MAS	830
84	MASST=MASST+ROUT*AREAT*G	MAS	840
85	MASSE=MASSE+ROUE*AREAE*G	MAS	850
86	THRUST=THRUST+ROUE2*AREAE	MAS	860
87	60 CONTINUE	MAS	870
88	RETURN	MAS	880
89	END	MAS	890

B.7 One-Dimensional Initialization (onedim.f)

OCR note: this listing received a conservative normalization pass for obvious token errors; unresolved ambiguities are tracked in conversion notes.

```

1 SUBROUTINE ONEDIM
2 C
3 C *****
4 C THIS SUBROUTINE CALCULATES THE 1-D INITIAL-DATA SURFACE
5 C *****
6 C
7 COMMON /AV/ IAV,CAV,NST,SMP,LSS,CTA,XMU,XLA,RKMU,QUT(81,21),QVT(81
8 1,21),QPT(81,21)
9 COMMON /ONESID/ UD(4),VD(4),PD(4),ROD(4)
10 COMMON /SOLUTIN/ U(81,21,2),V(81,21,2),P(81,21,2),RO(81,21,2)
11 COMMON /CNTRL/ LMAX,MMAX,NMAX,NPRINT,TCONV,FDT,GAMMA,RGAS,GAMI,GA
12 1M2,L1,L2,L3,M1,M2,DX,DY,DT,N,N1,N3,NASM,IVEL,ICHAR,N1D,LJET,JFLAG,
13 2IERR,IUI,IUO,DXR,DYR,LD,MD,LMD1,LMD3,IB,RSTAR,RSTARS,NPLOT,G,PC,TC
14 3,LC,PLOW,ROLOW
15 COMMON /GEMIRYC/ NGEOM,XI,RI,XT,RT,XE,RE,RCI,RCT,ANGI,ANGE,XW(81),
16 1YW(81),XWI(81),YWI(81),NXNY(81),NWPIS,IINT,IDIF,LT,NDIM
17 COMMON /GCB/ NGCB,XICB,RICB,XTCB,RTCB,XECB,RECB,RCICB,RCTCB,ANGICB
18 1,ANGEGB,XCB(81),YCB(81),XCB1(81),YCB1(81),NXNYCB(81),NCBPTS,IINTCB
19 2,IDIFCB,LECB
20 COMMON /BOC/ PT(21),TT(21),THETA(21),PE,MASSE,MASSI,MASST,THRUST,
21 1NSTAG
22 REAL MN3,NXNY,MASSI,MASST,NXNYCB,MASSE
23
24 MN3=0.01
25 IF (N1D,EQ,-1,OR,N1D,GT,2) MN3=2.0
26 GRGAS=1.0/(RGAS*G)
27 NXCK=0
28 ACOEF=2.0/(GAMMA+1.0)
29 BCOEF=(GAMMA-1.0)/(GAMMA+1.0)
30 CCOEF=(GAMMA+1.0)/2.0/(GAMMA-1.0)
31 IF (N1D,LT,0) GO TO 20
32 C
33 C OVERALL LOOP
34 C
35 IF (NGCB,NE,0) GO TO 10
36 RSTAR=RT
37 RSTARS=RT*RT
38 GO TO 20
39 10 RSTAR=YW(LT)-YCB(LT)
40 RSTARS=YW(LT)**2-YCB(LT)**2
41 20 DO 130 L=1,LMAX
42 IF (L,EQ,1,AND,N1D,EQ,-1) GO TO 130
43 IF (L,EQ,1,AND,N1D,GT,2) GO TO 130
44 X=XI+DX*(L-1)
45 IF (N1D,LT,0) GO TO 50
46 IF (NGCB,NE,0) GO TO 30
47 IF (X,LT,XT) GO TO 50
48 IF (X,GT,XT) GO TO 40
49 MN3=1.0
50 GO TO 100
51 30 IF (L,LT,LT) GO TO 50
52 IF (L,GT,LT) GO TO 40
53 MN3=1.0
54 GO TO 100
55 40 IF (NXCK,EQ,1) GO TO 50
56 IF (N1D,EQ,1,OR,N1D,EQ,3) MN3=1.1
57 IF (N1D,EQ,2,OR,N1D,EQ,4) MN3=0.9
58 NXCK=1
59 50 IF (NDIM,EQ,1) GO TO 60
60 RAD=YW(L)-YCB(L)
61 ARATIO=RAD/RSTAR
62 GO TO 70
63 60 RADS=YW(L)**2-YCB(L)**2
64 ARATIO=RADS/RSTARS
65 C
66 C NEWTON-RAPHSON ITERATION LOOP
67 C
68 70 DO 90 ITER=1,20
69 ABM = ACOEF + BCOEF * MN3**2

```

```

70  ABMC = ABM*CCOEF
71  FM = ABMC / MN3 - ARATIO
72  FPM = ABMC * (2.0 * BCOEF * CCOEF/ABM-1.0/MN3**2)
73  OMN3 = MN3
74  MN3 = OMN3 - FM/FPM
75  IF (MN3,GT,1.0,AND,OMN3,LT,1.0) MN3=0.99
76  IF (MN3,LT,1.0,AND,OMN3,GT,1.0) MN3=1.01
77  IF (MN3,GE,0.00) GO TO 80
78  MN3=-MN3
79  GO TO 90
80  IF (ABS(MN3-OMN3)/OMN3,LE,0.0005) GO TO 100
81 90  CONTINUE
82  PRINT 140, L
83  C
84  C  Fill IN 2-D ARRAYS LOOP
85  C
86 100 DEM = 1.0 + GAM2 * MN3 * MN3
87  DEMP = DEM*GAM1
88  DNXY = (NXNY(L) - NXNYCB(L)) / M1
89  DO 120 M=1,MMAX
90  P(L,M,1)=PT(M)/DEMP
91  TEMP=TT(M)/DEM
92  RO(L,M,1)=P(L,M,1)*GRGAS/TEMP
93  Q=MN3*SQRT(GAMMA*P(L,M,1)/RO(L,M,1))
94  DN=NXYCB(L)+DNXY*(M-1)
95  DNS=DN*DN
96  IF (DNS,EQ,0.0) GO TO 110
97  SIGN=1.0
98  IF (DN,GT,0.0) SIGN=-1.0
99  U(L,M,1)=Q/SQRT(1.0+DNS)
100 V(L,M,1)=SIGN*Q/SQRT(1.0+1.0/DNS)
101 GO TO 120
102 U(L,M,1)=Q
103 V(L,M,1)=0.0
104120 CONTINUE
105130 CONTINUE
106  RETURN
107  C
108 140 FORMAT (1H0,10X,93H***** THE 1-D SOLUTION FOR THE INITIAL-DATA SUR
109 1FACE FAILED TO CONVERGE IN 20 ITERATIONS AT L=,I2,6H *****
110  END

```

B.8 Centerbody Geometry (geomcb.f)

This subroutine calculates the centerbody radius and slope for cases with a centerbody configuration. It is used in the GEOM routine for geometry preprocessing. *OCR note: this listing was reconstructed from Appendix C PDF text blocks for readability; minor geometric/formatted-text normalization was required where scan quality was poor.*

```

1  SUBROUTINE GEOMCB
2  C
3  C  *****
4  C  THIS SUBROUTINE CALCULATES THE CENTERBODY RADIUS AND SLOPE
5  C  *****
6  C
7  C  OCR note:
8  C  Reconstructed from Appendix C listing text blocks. Some FORMAT
9  C  literal wording was normalized for readability in degraded regions.
10 C
11  DIMENSION YW(81),NXNY(81)
12  COMMON /AV/ IAV,CAV,NST,SMP,LSS,CTA,XMU,XLA,RKMU,QUT(81,21),QVT(81,21),QPT(81,21)
13  COMMON /ONESID/ UD(4),VD(4),PD(4),ROD(4)
14  COMMON /SOLUTN/ U(81,21,2),V(81,21,2),P(81,21,2),RO(81,21,2)
15  COMMON /CNTRLC/ LMAX,MMAX,NMAX,NPRINT,TCOINV,FDT,GAMMA,GRAS,GAM1,GAM2,
16  1L1,L2,L3,M1,M2,DX,DY,DT,N,N1,N3,NASM,IVEL,ICHAR,N1D,LJET,JFLAG,
17  2IERR,IUI,IUO,DXR,DYR,LD,MD,LMD1,LMD3,IB,RSTAR,RSTARS,NPLOT,G,PC,TC,
18  3LC,PLOW,ROLOW
19  COMMON /GCB/ NGCB,XICB,RICB,XTCB,RTCB,XECB,RECB,RCICB,RCTCB,ANGICB,
20  1ANGECB,XCB(81),YCB(81),XCBI(81),YCBI(81),NXNYCB(81),NCBPTS,IINTCB,

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```

21      2IDIFCB,LECB
22      COMMON /BCC/ PT(21),TT(21),THETA(21),PE,MASSE,MASSI,MASST,THRUST,
23      INSTAG
24      REAL MN3,NXNY,MASSI,MASST,NXNYCB,MASSE
25  C
26      GO TO (10,30,120,160),NGCB
27  C
28  C      CYLINDRICAL CENTERBODY CASE
29  C
30  10      IF (IU1.EQ.1) PRINT 230, XICB,RICB,XECB
31          IF (IU1.EQ.2) PRINT 240, XICB,RICB,XECB
32          LECB=LMAX
33          DO 20 L=1,LMAX
34              YCB(L)=RICB
35              NXNYCB(L)=0.0
36  20      CONTINUE
37          GO TO 200
38  C
39  C      CIRCULAR-ARC, CONICAL CENTERBODY CASE
40  C
41  30      XI=XICB
42          RI=RICB
43          RT=RTCB
44          XE=XECB
45          RCI=RCICB
46          RCT=RCTCB
47          ANGI=ANGICB
48          ANGE=ANGECB
49          RI=2.0*RT-RI
50          IF (RCI.EQ.0.0.OR.RCT.EQ.0.0) GO TO 190
51          ANI=ANGI*3.141593/180.0
52          ANE=ANGE*3.141593/180.0
53          XTAN=XI+RCI*SIN(ANI)
54          RTAN=RI+RCI*(COS(ANI)-1.0)
55          RT1=RT-RCT*(COS(ANI)-1.0)
56          XT1=XTAN+(RTAN-RT1)/TAN(ANI)
57          IF (XT1.GE.XTAN) GO TO 40
58          XT1=XTAN
59          RT1=RTAN
60  40      XT=XT1+RCT*SIN(ANI)
61          XTCE=XT
62          XT2=XT+RCT*SIN(ANE)
63          RT2=RT+RCT*(1.0-COS(ANE))
64          RE=RT2+(XE-XT2)*TAN(ANE)
65          RECB=RE
66          RI=2.0*RT-RI
67          RE=2.0*RT-RE
68          IF (IU1.EQ.1) PRINT 250, XI,RI,RT,XE,RCI,RCT,ANGI,ANGE,XT,RE
69          IF (IU1.EQ.2) PRINT 260, XI,RI,RT,XE,RCI,RCT,ANGI,ANGE,XT,RE
70          RI=2.0*RT-RI
71          RE=2.0*RT-RE
72          DO 110 L=1,LMAX
73              X=XI+(L-1)*DX
74              IF (X.GE.XI.AND.X.LE.XTAN) GO TO 50
75              IF (X.GT.XTAN.AND.X.LE.XT1) GO TO 60
76              IF (X.GT.XT1.AND.X.LE.XT) GO TO 70
77              IF (X.GT.XT.AND.X.LE.XT2) GO TO 80
78              IF (X.GT.XT2.AND.X.LE.XE) GO TO 90
79  C
80  50      YW(L)=RI+RCI*(COS(ASIN((X-XI)/RCI))-1.0)
81          NXNY(L)=(XI-X)/(YW(L)-RI+RCI)
82          GO TO 100
83  C
84  60      YW(L)=RT1+(XT1-X)*TAN(ANI)
85          NXNY(L)=TAN(ANI)
86          GO TO 100
87  C
88  70      YW(L)=RT+RCT*(1.0-COS(ASIN((XT-X)/RCT)))
89          NXNY(L)=(XT-X)/(RCT+RT-YW(L))
90          GO TO 100
91  C
92  80      YW(L)=RT+RCT*(1.0-COS(ASIN((X-XT)/RCT)))
93          NXNY(L)=(X-XT)/(RCT+RT-YW(L))
94          GO TO 100
95  C
96  90      YW(L)=RT2+(X-XT2)*TAN(ANE)

```

```

97      NXNY(L)=-TAN(ANE)
98  C
99 100   YCB(L)=2.0*RTCB-YW(L)
100     NXNYCB(L)=-NXNY(L)
101     IF (YCB(L).GE.0.0) GO TO 110
102     YCB(L)=0.0
103     NXNYCB(L)=0.0
104 110   CONTINUE
105     GO TO 200
106  C
107  C      GENERAL CENTERBODY CASE - INPUT CENTERBODY COORDINATES ONLY
108  C
109 120   PRINT 220
110     IF (IUI.EQ.1) PRINT 270, IINTCB,IDIFCB
111     IF (IUI.EQ.2) PRINT 280, IINTCB,IDIFCB
112     LDUM=NCBPTS
113     IF (LMAX.GT.NCBPTS) LDUM=LMAX
114     IP=1
115     DO 130 L=1,LMAX
116     XCB(L)=XICB+DX*(L-1)
117     CALL MILUP (XCB(L),YCB(L),IINTCB,NCBPTS,NCBPTS,1,IP,XCBI,YCBI)
118 130   CONTINUE
119     DO 150 L=1,LDUM
120     IF (L.GT.LMAX) GO TO 140
121     SLOPE=DIF(L,IDIFCB,LMAX,XCB,YCB)
122     NXNYCB(L)=-SLOPE
123     IF (YCB(L).GE.0.0) GO TO 135
124     YCB(L)=0.0
125     NXNYCB(L)=0.0
126     SLOPE=-NXNYCB(L)
127 135   IF (L.LE.NCBPTS.AND.L.LE.LMAX) PRINT 310, L,XCBI(L),YCBI(L),XCB(L),YCB(L),SLOPE
128     IF (L.GT.NCBPTS.AND.L.LE.LMAX) PRINT 320, L,XCB(L),YCB(L),SLOPE
129 140   IF (L.LE.NCBPTS.AND.L.GT.LMAX) PRINT 330, L,XCBI(L),YCBI(L)
130 150   CONTINUE
131     GO TO 200
132  C
133  C      GENERAL CENTERBODY CASE - INPUT CENTERBODY COORDINATES AND SLOPES
134  C
135 160   PRINT 220
136     IF (IUI.EQ.1) PRINT 290
137     IF (IUI.EQ.2) PRINT 300
138     DO 180 L=1,LMAX
139     XCB(L)=XICB+DX*(L-1)
140     IF (YCB(L).GE.0.0) GO TO 170
141     YCB(L)=0.0
142     NXNYCB(L)=0.0
143 170   SLOPE=-NXNYCB(L)
144     PRINT 340, L,XCB(L),YCB(L),SLOPE
145 180   CONTINUE
146     GO TO 200
147  C
148 190   PRINT 350
149     IERR=1
150     RETURN
151  C
152 200   IF (IUI.EQ.1) RETURN
153     DO 210 L=1,LMAX
154     YCB(L)=YCB(L)/2.54
155 210   CONTINUE
156     XICB=XICB/2.54
157     XECB=XECB/2.54
158     DX=DX/2.54
159     RETURN
160  C
161  C      FORMAT STATEMENTS
162  C
163 220   FORMAT (1H )
164 230   FORMAT (1H0,20X,52HA CYLINDRICAL CENTERBODY HAS BEEN SPECIFIED BY,
165     1XICB=,F8.4,12H (IN), RICB=,F8.4,16H (IN), AND XECB=,F8.4,5H (IN))
166 240   FORMAT (1H0,20X,52HA CYLINDRICAL CENTERBODY HAS BEEN SPECIFIED BY,
167     1XICB=,F8.4,12H (CM), RICB=,F8.4,16H (CM), AND XECB=,F8.4,5H (CM))
168 250   FORMAT (1H0,20X,62HA CIRCULAR-ARC, CONICAL CENTERBODY HAS BEEN SPECIFIED BY XICB=,F8.4,5H (IN),
169     17H, RICB=,F8.4,6H (IN),/,21X,8HRTCB=,F8.4,7H (IN),,5HXECB=,F8.4,5H (IN),
170     28H, RCICB=,F8.4,5H (IN),8H, RCTCB=,F8.4,5H (IN),9H, ANGICB=,F6.2,7H (DEG),
171     3/,21X,11HAND ANGECB=,F6.2,8H (DEG),,29HTHE COMPUTED VALUES ARE XTCB=,F8.4,
172     45H (IN),10H AND RECB=,F8.4,6H (IN),)

```



```

173 260 FORMAT (1H0,20X,62HA CIRCULAR-ARC, CONICAL CENTERBODY HAS BEEN SPECIFIED BY XICB=,F8.4,5H (CM) ,
174 17H, RICB=,F8.4,6H (CM) ,/,21X,8HRTCB=,F8.4,7H (CM) , ,5HXECB=,F8.4,5H (CM) ,
175 28H, RCICB=,F8.4,5H (CM) ,8H, RCTCB=,F8.4,5H (CM) ,9H, ANGICB=,F6.2,7H (DEG) ,
176 3/,21X,11HAND ANGECB=,F6.2,8H (DEG) , ,29HTHE COMPUTED VALUES ARE XTICB=,F8.4,
177 45H (CM) ,10H AND RECB=,F8.4,6H (CM) ,)
178 270 FORMAT (1H0,20X,76HA GENERAL CENTERBODY HAS BEEN SPECIFIED BY THE FOLLOWING PARAMETERS,
179 1 IINTCB=,11,9H, IDIFCB=,11,/,22X,1HL,10X,8HXCBI(IN) ,10X,8HYCBI(IN) ,9X,
180 27HXCBI(IN) ,10X,7HYCBI(IN) ,11X,5HSLOPE,/)
181 280 FORMAT (1H0,20X,76HA GENERAL CENTERBODY HAS BEEN SPECIFIED BY THE FOLLOWING PARAMETERS,
182 1 IINTCB=,11,9H, IDIFCB=,11,/,22X,1HL,10X,8HXCBI(CM) ,10X,8HYCBI(CM) ,9X,
183 27HXCBI(CM) ,10X,7HYCBI(CM) ,11X,5HSLOPE,/)
184 290 FORMAT (1H0,20X,68HA GENERAL CENTERBODY HAS BEEN SPECIFIED BY THE FOLLOWING PARAMETERS,
185 1 //,22X,1HL,11X,7HXCBI(IN) ,10X,7HYCBI(IN) ,11X,5HSLOPE,/)
186 300 FORMAT (1H0,20X,68HA GENERAL CENTERBODY HAS BEEN SPECIFIED BY THE FOLLOWING PARAMETERS,
187 1 //,22X,1HL,11X,7HXCBI(CM) ,10X,7HYCBI(CM) ,11X,5HSLOPE,/)
188 310 FORMAT (1H ,20X,I2,7X,F10.4,7X,F10.4,7X,F10.4,7X,F10.4,7X,F10.4)
189 320 FORMAT (1H ,20X,I2,41X,F10.4,7X,F10.4,7X,F10.4,7X,F10.4)
190 330 FORMAT (1H ,20X,I2,7X,F10.4,7X,F10.4)
191 340 FORMAT (1H ,20X,I2,7X,F10.4,7X,F10.4,7X,F10.4)
192 350 FORMAT (1H0,48H**** RCICB OR RCTCB WAS SPECIFIED AS ZERO ****)
193 END

```

B.9 Table Interpolation (mtlup.f)

This subroutine performs multilinear interpolation on tabular input data. It is called by GEOM to interpolate geometry and flow properties from tables. This routine was adapted from a NASA-Langley program. *OCR note: this listing was reconstructed from Appendix C PDF text blocks for readability; diagnostic/comment text literals were normalized where scan artifacts obscured wording.*

```

1 SUBROUTINE MTLUP (X,Y,M,N,MAX,NTAB,I,VARI,VARD)
2 C
3 C THIS SUBROUTINE IS CALLED BY SUBROUTINE GEOM TO INTERPOLATE FOR
4 C EQUALLY SPACED NOZZLE WALL COORDINATES FOR THE TABULAR INPUT CASE.
5 C SUBROUTINE MTLUP WAS TAKEN FROM THE NASA-LANGLEY PROGRAM LIBRARY.
6 C
7 C MODIFICATION OF LIBRARY INTERPOLATION SUBROUTINE FTLUP
8 C MULTIPLE TABLE LOOK-UP ON ONE INDEPENDENT VARIABLE TABLE
9 C USES AN EXTERNAL INTERVAL POINTER (I) TO START SEARCH
10 C I LESS THAN 0 WILL CHECK MONOTONICITY
11 C
12 C OCR note:
13 C Reconstructed from Appendix C listing text blocks. Some original
14 C diagnostic text literals were normalized where the scan was unreadable.
15 C Diagnostic FORMAT wording is normalized for readability.
16 C
17 DIMENSION VARI(N),VARD(MAX,1),Y(1),V(3),YY(2)
18 LOGICAL EX
19 C
20 IF (M.EQ.0) GO TO 170
21 IF (N.LE.1) GO TO 170
22 EX=.FALSE.
23 IF (I.GE.0) GO TO 60
24 IF (N.LT.2) GO TO 60
25 C
26 C MONOTONICITY CHECK
27 C
28 IF (VARI(2)-VARI(1)) 20,20,40
29 C
30 C ERROR IN MONOTONICITY
31 C
32 10 PRINT 190, N
33 STOP
34 C
35 C MONOTONIC DECREASING
36 C
37 20 DO 30 J=2,N
38 IF (VARI(J)-VARI(J-1)) 30,10,10
39 30 CONTINUE
40 GO TO 60

```

```

41 C
42 C    MONOTONIC INCREASING
43 C
44 40    DO 50 J=2,N
45      IF (VARI(J)-VARI(J-1)) 10,10,50
46 50    CONTINUE
47 C
48 C    INTERPOLATION
49 C
50 60    IF (I.LE.0) I=1
51      IF (I.GE.N) I=N-1
52 C
53 C    LOCATE I INTERVAL (VARI(I) .LE. X .LT. VARI(I+1))
54 C
55      IF ((VARI(I)-X)*(VARI(I+1)-X)) 100,100,70
56 C
57 C    IN GIVES DIRECTION FOR SEARCH OF INTERVALS
58 C
59 70    IN=SIGN(1.0,(VARI(N)-VARI(I))*(X-VARI(I)))
60 C
61 C    IF X OUTSIDE ENDPOINTS, EXTRAPOLATE FROM END INTERVAL
62 C
63 80    IF ((I+IN).LE.0) GO TO 90
64      IF ((I+IN).GE.N) GO TO 90
65      I=I+IN
66      IF ((VARI(I)-X)*(VARI(I+1)-X)) 100,100,80
67 C
68 C    EXTRAPOLATION
69 C
70 90    EX=.TRUE.
71 100   IF (M.EQ.2) GO TO 120
72 C
73 C    FIRST ORDER
74 C
75      DO 110 NT=1,NTAB
76 110   Y(NT)=(VARD(I,NT)*(VARI(I+1)-X)-VARD(I+1,NT)*(VARI(I)-X))/
77       1 (VARI(I+1)-VARI(I))
78       IF (EX) I=I-IN
79       RETURN
80 C
81 C    SECOND ORDER
82 C
83 120   IF (N.EQ.2) GO TO 110
84       IF (I.EQ.(N-1)) GO TO 140
85       IF (I.EQ.1) GO TO 130
86 C
87 C    PICK THIRD POINT
88 C
89       SK=VARI(I+1)-VARI(I)
90       IF (ABS(X-VARI(I+1)).LT.ABS(VARI(I+2)-X)) GO TO 140
91 130   L=I
92       GO TO 150
93 140   L=I-1
94 150   V(1)=VARI(L)-X
95       V(2)=VARI(L+1)-X
96       V(3)=VARI(L+2)-X
97       DO 160 NT=1,NTAB
98       YY(1)=(VARD(L,NT)*V(2)-VARD(L+1,NT)*V(1))/(VARI(L+1)-VARI(L))
99       YY(2)=(VARD(L+1,NT)*V(3)-VARD(L+2,NT)*V(2))/(VARI(L+2)-VARI(L+1))
100 160  Y(NT)=(YY(1)*V(3)-YY(2)*V(1))/(VARI(L+2)-VARI(L))
101       IF (EX) I=I-IN
102       RETURN
103 C
104 C    ZERO ORDER
105 C
106 170   DO 180 NT=1,NTAB
107 180   Y(NT)=VARD(I,NT)
108       RETURN
109 C
110 190   FORMAT (1H1,49H TABLE BELOW OUT OF ORDER FOR MITLUP, N = ,15)
111   END

```

B.10 Numerical Differentiation (diff.f)

This function calculates numerical derivatives for tabular input data. It is used by GEOM to compute nozzle wall slopes for the tabular input case, providing finite-difference approximations to derivatives. *OCR note: this listing was reconstructed from Appendix C PDF text blocks for readability; comments were normalized while preserving the derivative computation logic.*

```

1      FUNCTION DIF (L,M,NP,VARI,VARD)
2      C
3      C      THIS FUNCTION IS CALLED BY SUBROUTINE GEOM TO CALCULATE THE
4      C      NOZZLE WALL SLOPE FOR THE TABULAR INPUT CASE. FUNCTION DIF WAS
5      C      TAKEN FROM THE NASA-LANGLEY PROGRAM LIBRARY.
6      C
7      C      THIS FUNCTION SUBPROGRAM FINDS THE DERIVATIVE AT A GIVEN POINT L
8      C      FOR THE DESIRED X AND Y IN A GIVEN TABLE. THE N-POINT LAGRANGIAN
9      C      FORMULA IS USED WHERE N IS ODD.
10     C
11     C      L      = INTEGER, POINT OF X AND Y AT WHICH DERIVATIVE IS FOUND
12     C      M      = INTEGER, 1-5, TO DETERMINE THE POINT FORMULA, N=2*M+1
13     C      NP     = INTEGER, NUMBER OF POINTS IN TABLE OF VARIABLES
14     C      VARI   = ARRAY OF INDEPENDENT VARIABLE, X, VARI(NP)
15     C      VARD   = ARRAY OF DEPENDENT VARIABLE, Y, VARD(NP)
16     C
17     C      OCR note:
18     C      Reconstructed from Appendix C listing text blocks; numeric logic
19     C      preserved and comments normalized for readability.
20     C
21     DIMENSION VARI(NP),VARD(NP),X(11),Y(11)
22     C
23     DIF=1.777E19
24     IF (M.LT.1) RETURN
25     N=2*M+1
26     IF (M.GT.5.OR.N.GT.NP) RETURN
27     M1=M+1
28     M2=NP-M1
29     K=L
30     IF (L.LE.M1.OR.N.EQ.NP) GO TO 10
31     K=M1
32     IF (L.LT.M2) GO TO 10
33     K=L-(NP-N)
34 10    MX=L-K
35     DO 20 J=1,N
36     MJ=MX+J
37     X(J)=VARI(MJ)
38 20    Y(J)=VARD(MJ)
39     A=1.0
40     B=0.0
41     C=0.0
42     DO 40 J=1,N
43     IF (J.EQ.K) GO TO 40
44     P=1.0
45     DO 30 I=1,N
46     IF (I.EQ.J) GO TO 30
47     P=P*(X(J)-X(I))
48 30    CONTINUE
49     T=X(K)-X(J)
50     B=B+Y(J)/(P*T)
51     A=A*T
52     C=C+1.0/T
53 40    CONTINUE
54     DIF=A*B+Y(K)*C
55     RETURN
56     END

```

B.11 Mapping Functions (map.f)

This subroutine calculates coordinate mapping functions, converting between physical (x, y) coordinates and transformed (ξ, η, ζ) computational coordinates used in the finite-difference scheme. *OCR note: this listing was replaced with a cleaner in-repo transcription and is currently the most readable available version.*

1	SUBROUTINE MAP(IP, L, M, AL, BE, DE, LD1, AL1, BE1, DE1)	MAP	10
2	C	MAP	20
3	C *****	MAP	30
4	C	MAP	40
5	C THIS SUBROUTINE CALCULATES THE MAPPING FUNCTIONS	MAP	50
6	C	MAP	60
7	C *****	MAP	70
8	C	MAP	80
9	COMMON /AV/ IAV,CAV,NST,SMP,LSS,CTA,XMU,XLA,RKMU,QUT(81,21),QVT(81,21)	MAP	90
10	1,21),QPT(81,21)	MAP	100
11	COMMON /ONESID/ UD(4),VD(4),PD(4),ROD(4)	MAP	110
12	COMMON /SOLUTN/ U(81,21,2),V(81,21,2),P(81,21,2),RO(81,21,2)	MAP	120
13	COMMON /CNTRLC/ LMAX,MMAX,NMAX,NPRINT,TOONV,FTD,GAMMA,RGAS,GAMI,GAMAP	MAP	130
14	1M2,L1,L2,L3,M1,M2,DX,DY,DT,N,N1,N3,NASM,IVEL,ICHAR,NID,LJET,JFLAG,MAP	MAP	140
15	2IERR,IUI,IUO,DXR,DYR,LD,MD,LMND,LMD3,IB,RSTAR,RSTARS,NPLOT,G,PC,TCMAP	MAP	150
16	3,LC,FLOW,ROLOW	MAP	160
17	COMMON /GEMIRYC/ NGEOM,XI,RI,XT,RT,XE,RE,RCI,RCT,ANGI,ANGE,XW(81),	MAP	170
18	1YW(81),XWI(81),YWI(81),NXNY(81),NWPTS,IINT,IDIF,LT,NDIM	MAP	180
19	COMMON /GCB/ NGCB,XICB,RI CB,XTCB,RTCB,XECB,RECB,RCICB,RCTCB,ANGICB,MAP	MAP	190
20	2,ANGE CB,XCB(81),YCB(81),XCBI(81),YCB(81),NXNYCB(81),NCBPTS,IINTCB,MAP	MAP	200
21	3,IDI FC B,LECB	MAP	210
22	COMMON /BOC/ PT(21),TT(21),THETA(21),PE,MASSE,MASSI,MASST,THRUST,NMAP	MAP	220
23	1STAG	MAP	230
24	REAL MN3,NXNY,MASSI,MASST,NXNYCB,MASSE	MAP	240
25	C	MAP	250
26	BE=1.0/(YW(L)-YCB(L))	MAP	260
27	IF (IP,EQ,0) RETURN	MAP	270
28	Y=(M-1)*DY	MAP	280
29	AL=BE*(NXNYCB(L)+Y*(NXNY(L)-NXNYCB(L)))	MAP	290
30	DE=-BE*Y*XWI(L)	MAP	300
31	IF (IP,EQ,1) RETURN	MAP	310
32	BE1=1.0/(YW(LD1)-YCB(LD1))	MAP	320
33	AL1=BE1*(NXNYCB(LD1)+Y*(NXNY(LD1)-NXNYCB(LD1)))	MAP	330
34	DE1=-BE1*Y*XWI(LD1)	MAP	340
35	RETURN	MAP	350
36	END	MAP	360

B.12 Artificial Viscosity (shock.f)

This subroutine calculates local artificial viscosity terms used to stabilize the solution across shock waves and steep gradients. Following the method of Lax and Wendroff, it provides controlled numerical dissipation to prevent oscillations. *OCR note: this listing was reconstructed from Appendix C PDF text blocks for readability and aligned to Appendix C line blocks in the viscosity-energy term region.*

1	SUBROUTINE SHOCK(IPASS)
2	C
3	C *****
4	C THIS SUBROUTINE CALCULATES THE LOCAL ARTIFICIAL VISCOSITY TERMS
5	C FOR SHOCK COMPUTATIONS
6	C *****
7	C
8	C OCR note:
9	C Reconstructed from Appendix C PDF text blocks and aligned to
10	C Appendix C line blocks for the viscosity-energy term structure.
11	C
12	COMMON /AV/ IAV,CAV,NST,SMP,LSS,CTA,XMU,XLA,RKMU,QUT(81,21),QVT(81,21),QPT(81,21)

```

13 COMMON /ONESID/ UD(4),VD(4),PD(4),ROD(4)
14 COMMON /SOLUTN/ U(81,21,2),V(81,21,2),P(81,21,2),RO(81,21,2)
15 COMMON /CNTRL/ LMAX,MMAX,NMAX,NPRINT,TCONV,FDT,GAMMA,RGAS,GAM1,GAM2,
16 1L1,L2,L3,M1,M2,DX,DY,DT,N,N1,N3,NASM,IVEL,ICHAR,NID,LJET,JFLAG,
17 2IEER,IUI,IUO,DXR,DYR,LD,MD,LMD1,LMD3,IB,RSTAR,RSTARS,NPLOT,G,PC,TC,
18 3LC,FLOW,ROLOW
19 COMMON /GEMTRYC/ NGEOM,XI,RI,XT,RT,XE,RE,RCI,RCT,ANGI,ANGE,XW(81),
20 YW(81),XWI(81),YWI(81),NXNY(81),NWPTS,IINT,IDIF,LT,NDIM
21 COMMON /GCB/ NGCB,XICB,RICB,XTCB,RTCB,XECB,RECB,RCICB,RCTCB,ANGICB,
22 1ANGECB,XCB(81),YCB(81),XCB(81),YCB(81),NXNYCB(81),NCBPTS,IINTCB,
23 2IDIFCB,LECB
24 COMMON /BOC/ PT(21),TT(21),THETA(21),PE,MASSE,MASSI,MASST,THRUST,
25 INSTAG
26 REAL MN3,NXNY,MASSI,MASST,NXNYCB,MASSE
27 C
28 GO TO (10,160),IPASS
29 C
30 C CALCULATE LOCAL ARTIFICAL VISCOSITY FOR SHOCK COMPUTATIONS
31 C
32 10 IF (N.NE.1) GO TO 30
33 NC=0
34 RG=RGAS*G
35 CTA1=1.0-CTA
36 DO 20 L=1,LMAX
37 DO 20 M=1,MMAX
38 QUT(L,M)=0.0
39 QVT(L,M)=0.0
40 QPT(L,M)=0.0
41 20 CONTINUE
42 30 RDUM=CAV*DT*DX*DY*2.0
43 NC=NC+1
44 NLINE=0
45 IF (NC.NE.NPRINT) GO TO 40
46 PRINT 200
47 PRINT 190,N,PC
48 C
49 40 DO 150 L=LSS,L1
50 DO 150 M=1,MMAX
51 LMD2=LD*(M-1)+LMD1
52 LMMD2=LD*(M-2)+LMD1
53 LMPD2=LD*M+LMD1
54 LMN1=L+LMD2
55 LP=L+1+LMD2
56 LM=L-1+LMD2
57 MP=L+LMPD2
58 MM=L+LMMD2
59 LPM=LP+1+LMPD2
60 LPM=LP+1+LMMD2
61 LMM=LM-1+LMPD2
62 LMM=LM-1+LMMD2
63 CALL MAP (1,L,M,AL,BE,DE,LD1,AL1,BE1,DE1)
64 C
65 C CHECK TO SEE IF THE DIVERGENCE OF THE VELOCITY IS NEGATIVE
66 C
67 UX=0.5*(U(LP)-U(LM))*DXR
68 IF (M.EQ.1) GO TO 50
69 IF (M.EQ.MMAX) GO TO 60
70 UY=0.5*(U(MP)-U(MM))*DYR
71 VY=0.5*(V(MP)-V(MM))*DYR
72 GO TO 70
73 50 UY=(U(MP)-U(LMN1))*DYR
74 VY=(V(MP)-V(LMN1))*DYR
75 GO TO 70
76 60 UY=(U(LMN1)-U(MM))*DYR
77 VY=(V(LMN1)-V(MM))*DYR
78 70 DIV=UX+AL*UY+BE*VY
79 IF (DIV.LT.0.0) GO TO 80
80 QUT(L,M)=0.0
81 QVT(L,M)=0.0
82 QPT(L,M)=0.0
83 GO TO 150
84 C
85 80 OQUT=QUT(L,M)
86 OQVT=QVT(L,M)
87 OQPT=QPT(L,M)
88

```

```

89    UX1=(U(LMN1)-U(LM)) *DXR
90    UX2=(U(LP)-U(LMN1)) *DXR
91    VX1=(V(LMN1)-V(LM)) *DXR
92    VX2=(V(LP)-V(LMN1)) *DXR
93    TM=P(LM) / (RO(LM) *RG)
94    T=P(LMN1) / (RO(LMN1) *RG)
95    TP=P(LP) / (RO(LP) *RG)
96    TX1=(T-TM) *DXR
97    TX2=(TP-T) *DXR
98
99    LDUM=L-1
100   CALL MAP (1,LDUM,M,ALM,BEM,DE,LD1,AL1,BE1,DE1)
101   LDUM=L+1
102   CALL MAP (1,LDUM,M,ALP,BEP,DE,LD1,AL1,BE1,DE1)
103   BE1=0.5*(BEM+BE)
104   BE2=0.5*(BEP+BE)
105   AL1=0.5*(ALM+AL)
106   AL2=0.5*(ALP+AL)
107
108   IF (M.EQ.1) GO TO 90
109   IF (M.EQ.MMAX) GO TO 100
110 C
111 C   CALCULATE THE INTERIOR POINT QUANTITIES
112 C
113   UY1=0.25*(U(MP)+U(LMMP)-U(MM)-U(LMMM)) *DYR
114   UY2=0.25*(U(MP)+U(LPMP)-U(MM)-U(LPMM)) *DYR
115   VY1=0.25*(V(MP)+V(LMMP)-V(MM)-V(LMMM)) *DYR
116   VY2=0.25*(V(MP)+V(LPMP)-V(MM)-V(LPMM)) *DYR
117   UX3=0.25*(U(LP)+U(LPMM)-U(LM)-U(LMMM)) *DXR
118   UX4=0.25*(U(LP)+U(LPMP)-U(LM)-U(LMMP)) *DXR
119   VX3=0.25*(V(LP)+V(LPMM)-V(LM)-V(LMMM)) *DXR
120   VX4=0.25*(V(LP)+V(LPMP)-V(LM)-V(LMMP)) *DXR
121   VY3=(V(LMN1)-V(MM)) *DYR
122   VY4=(V(MP)-V(LMN1)) *DYR
123   UY3=(U(LMN1)-U(MM)) *DYR
124   UY4=(U(MP)-U(LMN1)) *DYR
125
126   TMY=P(MM) / (RO(MM) *RG)
127   TPY=P(MP) / (RO(MP) *RG)
128   TMM=P(LMMM) / (RO(LMMM) *RG)
129   TMP=P(LMMP) / (RO(LMMP) *RG)
130   TPM=P(LPMM) / (RO(LPMM) *RG)
131   TPP=P(LPMP) / (RO(LPMP) *RG)
132   TY1=0.25*(TPY+TMP-TMY-TMM) *DYR
133   TY2=0.25*(TPP+TPY-TPM-TMY) *DYR
134   TX3=0.25*(TP+TPM-TM-TMM) *DXR
135   TX4=0.25*(TPP+TP-TMP-TM) *DXR
136   TY3=(T-TMY) *DYR
137   TY4=(TPY-T) *DYR
138
139   MDUM=M-1
140   CALL MAP (1,L,MDUM,ALMY,BEMY,DE,LD1,AL1,BE1,DE1)
141   MDUM=M+1
142   CALL MAP (1,L,MDUM,ALPY,BEPY,DE,LD1,AL1,BE1,DE1)
143   BE3=0.5*(BEMY+BE)
144   BE4=0.5*(BEPY+BE)
145   AL3=0.5*(ALMY+AL)
146   AL4=0.5*(ALPY+AL)
147   GO TO 110
148 C
149 C   CALCULATE THE CENTERLINE POINT QUANTITIES
150 C
151 90   UY1=0.5*(U(MP)+U(LMMP)-U(LMN1)-U(LM)) *DYR
152   VY1=0.5*(V(MP)+V(LMMP)-V(LMN1)-V(LM)) *DYR
153   UY2=0.5*(U(LPMP)+U(MP)-U(LP)-U(LMN1)) *DYR
154   VY2=0.5*(V(LPMP)+V(MP)-V(LP)-V(LMN1)) *DYR
155   UX3=0.0
156   VX3=0.0
157   UX4=0.0
158   VX4=0.0
159   THEW=ATAN(-NXNYCB(L))
160   THE=ATAN(V(MP)/U(MP))
161   VMAG=SQRT(U(MP)*U(MP)+V(MP)*V(MP))
162   RTHE=2.0*THEW-THE
163   UR=VMAG*COS(RTHE)
164   VR=VMAG*SIN(RTHE)

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```

165      UY3=(U(LMN1)-UR)*DYZ
166      VY3=(V(LMN1)-VR)*DYZ
167      UY4=(U(MP)-U(LMN1))*DYZ
168      VY4=(V(MP)-V(LMN1))*DYZ
169      TPY=P(MP)/(RO(MP)*RG)
170      TMP=P(LMP)/(RO(LMP)*RG)
171      TPP=P(LPMP)/(RO(LPMP)*RG)
172      TY1=0.0
173      TY2=0.0
174      TX3=0.25*(TPP+TP-TMP-TM)*DXR
175      TX4=TX3
176      TY4=(TPY-T)*DYZ
177      TY3=-TY4
178      BE3=BE
179      AL3=AL
180      MDUM=M+1
181      CALL MAP (1,L,MDUM,AL4,BE4,DE,LD1,AL1,BE1,DE1)
182      GO TO 110
183 C
184 C   CALCULATE THE WALL POINT QUANTITIES
185 C
186 100  UY1=0.5*(U(LMN1)+U(LM)-U(MM)-U(MMM))*DYZ
187      VY1=0.5*(V(LMN1)+V(LM)-V(MM)-V(MMM))*DYZ
188      UY2=0.5*(U(LP)+U(LMN1)-U(LPMM)-U(MM))*DYZ
189      VY2=0.5*(V(LP)+V(LMN1)-V(LPMM)-V(MM))*DYZ
190      UX3=0.0
191      VX3=0.0
192      UX4=0.0
193      VX4=0.0
194      UY3=(U(LMN1)-U(MM))*DYZ
195      VY3=(V(LMN1)-V(MM))*DYZ
196      THEW=ATAN(-NXNY(L))
197      THE=ATAN(V(MM)/U(MM))
198      VMAG=SQRT(U(MM)*U(MM)+V(MM)*V(MM))
199      RTHE=2.0*THEW-THE
200      UR=VMAG*COS(RTHE)
201      VR=VMAG*SIN(RTHE)
202      UY4=(UR-U(LMN1))*DYZ
203      VY4=(VR-V(LMN1))*DYZ
204      TPM=P(LPMM)/(RO(LPMM)*RG)
205      TMY=P(MM)/(RO(MM)*RG)
206      TMM=P(MMM)/(RO(MMM)*RG)
207      TY1=0.0
208      TY2=0.0
209      TX3=0.25*(TP+TPM-TM-TMM)*DXR
210      TX4=TX3
211      TY3=(T-TMY)*DYZ
212      TY4=-TY3
213      MDUM=M-1
214      CALL MAP (1,L,MDUM,AL3,BE3,DE,LD1,AL1,BE1,DE1)
215      BE4=BE
216      AL4=AL
217 C
218 110  UXY1=UX1+AL1*UY1
219      UXY2=UX2+AL2*UY2
220      UXY3=UX3+AL3*UY3
221      UXY4=UX4+AL4*UY4
222      UXY12=0.5*(UX1+UX2+AL3*UY3+AL4*UY4)
223
224      VXY1=VX1+AL1*VY1
225      VXY2=VX2+AL2*VY2
226      VXY3=VX3+AL3*VY3
227      VXY4=VX4+AL4*VY4
228      VXY12=0.5*(VX1+VX2+AL3*VY3+AL4*VY4)
229
230      BUY1=BE1*UY1
231      BUY2=BE2*UY2
232      BUY3=BE3*UY3
233      BUY4=BE4*UY4
234      BUY34=0.5*(BUY3+BUY4)
235
236      BVY1=BE1*VY1
237      BVY2=BE2*VY2
238      BVY3=BE3*VY3
239      BVY4=BE4*VY4
240      BVY34=0.5*(BVY3+Bvy4)

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```

241
242     TXY1=TX1+AL1*TY1
243     TXY2=TX2+AL2*TY2
244     TXY3=TX3+AL3*TY3
245     TXY4=TX4+AL4*TY4
246 C
247 C     CALCULATE THE ARTIFICIAL VISCOSITY COEFFICIENTS
248 C
249     DIV=UXY12+BVY34
250     VID=VXY12-BUY34
251     RLA=XLA*RDUM*ABS(DIV)/BE
252     RMU=XMU*RDUM*ABS(VID)/BE
253     RK=RMU*GAM1*RG/RKMU
254     RLP2M=RLA+2.0*RMU
255     RLA2=2.0*RLA
256     RMU2=2.0*RMU
257     RLPM=RLA+RMU
258     UVTA=0.0
259     VVTA=0.0
260     PVTa=0.0
261     PCTA=0.0
262
263     IF (NDIM.EQ.0) GO TO 130
264 C
265 C     CALCULATE THE AXISYMMETRIC TERMS
266 C
267     IF (M.EQ.1.AND.YCB(L).EQ.0.0) GO TO 120
268     Y=FLOAT(M-1)*DY/BE+YCB(L)
269     VB=V(LMN1)
270     UVTA=(RLPM*VXY12+RMU*BUY34)/Y
271     VVTA=RLP2M*(BVY34+VB/Y)/Y
272     PVTa=(RLP2M*VB*VB/Y+RLA2*VB*(BVY34+UXY12))/Y
273     PCTA=RK*0.5*(BE4*TY4+BE3*TY3)/Y
274     DUVTA=RLPM*BE*(VXY4-VXY3)*DYR+RMU*BE*(BUY4-BUY3)*DYR
275     DVVTA=RLP2M*0.5*BE*(BVY4-BVY3)*DYR
276     DPVTA=(RLP2M+RLA2)*BVY34+BVY34+RLA2*BVY34*UXY12
277     DPCTA=RK*BE*(BE4*TY4-BE3*TY3)*DYR
278     IF (ABS(UVTA).GT.ABS(DUVTA)) UVTA=DUVTA
279     IF (ABS(VVTA).GT.ABS(DVVTA)) VVTA=DVVTA
280     IF (ABS(PVTa).GT.ABS(DPVTA)) PVTa=DPVTA
281     IF (ABS(PCTA).GT.ABS(DPCTA)) PCTA=DPCTA
282     GO TO 130
283 120 UVTA=RLPM*BE*(VXY4-VXY3)*DYR+RMU*BE*(BUY4-BUY3)*DYR
284     VVTA=RLP2M*0.5*BE*(BVY4-BVY3)*DYR
285     PVTa=(RLP2M+RLA2)*BVY34+BVY34+RLA2*BVY34*UXY12
286     PCTA=RK*BE*(BE4*TY4-BE3*TY3)*DYR
287 C
288 C     CALCULATE THE ARTIFICIAL VISCOSITY TERMS
289 C
290 130 QUT(L,M)=RLP2M*(UXY2-UXY1)*DXR
291     1 + AL*(RLP2M*(UXY4-UXY3)+RLA*(BVY4-BVY3))*DYR
292     2 + RMU*BE*(VXY4-BUY4-VXY3+BUY3)*DYR + UVTA
293
294     QVT(L,M)=RMU*(VXY2-BUY2-VXY1+BUY1)*DXR
295     1 + RMU*AL*(VXY4-BUY4-VXY3+BUY3)*DYR
296     2 + BE*(RLA*(UXY4-UXY3)+RLP2M*(BVY4-BVY3))*DYR + VVTA
297
298     QPT(L,M)=RO(LMN1)*(GAMMA-1.0)*
299     1 (RLP2M*(UXY12*UXY12+BVY34*BVY34)
300     2 + RMU*(VXY12*VXY12+BUY34*BUY34)
301     3 + RLA2*UXY12*BVY34 + RMU2*BUY34*VXY12
302     4 + RK*((TXY2-TXY1)*DXR + AL*(TXY4-TXY3)*DYR
303     5 + BE*(BE4*TY4-BE3*TY3)*DYR) + PVTa + PCTA)
304
305     QUT(L,M)=CTA*QUT(L,M)+CTA1*QUT
306     QVT(L,M)=CTA*QVT(L,M)+CTA1*QVT
307     QPT(L,M)=CTA*QPT(L,M)+CTA1*QPT
308
309     IF (NC.NE.NPRINT) GO TO 150
310     NLINE=NLINE+1
311     IF (NLINE.LT.55) GO TO 140
312     PRINT 200
313     PRINT 190, N, PC
314     NLINE=1
315 140 QPT(L,M)=QPT(L,M)/PC
316     PRINT 180, L,M,QUT(L,M),QVT(L,M),QPT(L,M)

```



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317      QPT(L,M)=QPT(L,M)*PC
318 150  CONTINUE
319
320      IF (NC.EQ.NPRINT) NC=0
321      RETURN
322 C
323 C      SMOOTH THE FLOW VARIABLES IF REQUESTED
324 C
325 160  IF (SMP.LT.0.0.OR.SMP.GE.1.0) RETURN
326      SMP4=0.25*(1.0-SMP)
327      DO 170 L=2,L1
328          U(L,MMAX,N3)=SMP4*(U(L-1,MMAX,N3)+U(L+1,MMAX,N3)+2.0*U(L,M1,N3))
329          1 +SMP*U(L,MMAX,N3)
330          V(L,MMAX,N3)=-U(L,MMAX,N3)*NXNY(L)
331          P(L,MMAX,N3)=SMP4*(P(L-1,MMAX,N3)+P(L+1,MMAX,N3)+2.0*P(L,M1,N3))
332          1 +SMP*P(L,MMAX,N3)
333          RO(L,MMAX,N3)=SMP4*(RO(L-1,MMAX,N3)+RO(L+1,MMAX,N3)+2.0*RO(L,M1,N3))
334          1 +SMP*RO(L,MMAX,N3)
335
336          U(L,1,N3)=SMP4*(U(L-1,1,N3)+U(L+1,1,N3)+2.0*U(L,2,N3))
337          1 +SMP*U(L,1,N3)
338          V(L,1,N3)=-U(L,1,N3)*NXNYCB(L)
339          P(L,1,N3)=SMP4*(P(L-1,1,N3)+P(L+1,1,N3)+2.0*P(L,2,N3))
340          1 +SMP*P(L,1,N3)
341          RO(L,1,N3)=SMP4*(RO(L-1,1,N3)+RO(L+1,1,N3)+2.0*RO(L,2,N3))
342          1 +SMP*RO(L,1,N3)
343
344      DO 170 M=2,M1
345          LMD2=LD*(M-1)+LMD3
346          LMMD2=LD*(M-2)+LMD3
347          LMPD2=LD*M+LMD3
348          LMN3=L+LMD2
349          LP=L+1+LMD2
350          LM=L-1+LMD2
351          MP=L+LMPD2
352          MM=L+LMMD2
353          U(LMN3)=SMP4*(U(LM)+U(LP)+U(MM)+U(MP))+SMP*U(LMN3)
354          V(LMN3)=SMP4*(V(LM)+V(LP)+V(MM)+V(MP))+SMP*V(LMN3)
355          P(LMN3)=SMP4*(P(LM)+P(LP)+P(MM)+P(MP))+SMP*P(LMN3)
356          RO(LMN3)=SMP4*(RO(LM)+RO(LP)+RO(MM)+RO(MP))+SMP*RO(LMN3)
357 170  CONTINUE
358
359      RETURN
360 180  FORMAT (1H ,5X,2I5,3F18.8)
361 190  FORMAT (1H0,63HLOCAL ARTIFICIAL VISCOSITY PARAMETERS FOR SHOCK CALC,
362          1ULATIONS, N=,14,5H (PC=,F5.1,1H) ,/,/,10X,1HL,4X,1HM,10X,3HQUT,
363          2 11X,3HQVT,11X,3HQPT,/)
364 200  FORMAT (1H1)
365      END

```

B.13 Exit Boundary Conditions (exitt.f)

This subroutine calculates boundary mesh points at the nozzle exit. It implements the exit boundary condition calculation based on prescribed exit plane conditions (pressure, or characteristic relations for subsonic flow). *OCR note: this listing was reconstructed from Appendix C PDF text blocks for readability and line-aligned to the degraded Appendix C compatibility-equation block.*

```

1      SUBROUTINE EXITT
2 C
3 C      *****
4 C      THIS SUBROUTINE CALCULATES THE BOUNDARY MESH POINTS AT THE
5 C      NOZZLE EXIT FOR SUBSONIC FLOW
6 C      *****
7 C
8 C      OCR note:
9 C      This file was reconstructed from Appendix C PDF text blocks.
10 C      Compatibility-equation updates were line-aligned to Appendix C
11 C      degraded text blocks and normalized to fixed-form syntax.
12 C

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13 COMMON /AV/ IAV,CAV,NST,SMP,LSS,CTA,XMU,XLA,RKMU,QUT(81,21),QVT(81,21),QPT(81,21)
14 COMMON /ONESID/ UD(4),VD(4),PD(4),ROD(4)
15 COMMON /SOLUTN/ U(81,21,2),V(81,21,2),P(81,21,2),RO(81,21,2)
16 COMMON /CNTRLC/ LMAX,MMAX,NMAX,NPRINT,TOONV,FDT,GAMMA,RGAS,GAM1,GAM2,
17 L1,L2,L3,M1,M2,DX,DY,DT,N,N1,N3,NASM,IVEL,ICHAR,N1D,LJET,JFLAG,
18 2IERR,IUI,IUO,DXR,DYR,LD,MD,LMD1,LMD3,IB,RSTAR,RSTARS,NPLOT,G,PC,TC,
19 3LC,FLOW,ROLOW
20 COMMON /GEMIRYC/ NGBOM,XI,RI,XT,RT,XE,RE,RCI,RCT,ANGI,ANGE,XW(81),
21 1YW(81),XWI(81),YWI(81),NXNY(81),NWPTS,IINT,IDIF,LT,NDIM
22 COMMON /GCB/ NGCB,XICB,RICB,XTCB,RTCB,XECB,RECB,RCICB,RCTCB,ANGICB,
23 1ANGECB,XCB(81),YCB(81),XCB1(81),YCB1(81),NXNYCB(81),NCBPTS,IINTCB,
24 2DIFCB,LECB
25 COMMON /BOCC/ PT(21),TT(21),THETA(21),PE,MASSE,MASSI,MASST,THRUST,
26 1NSTAG
27 REAL MN3,NXNY,MASSI,MASST,NXNYCB,MASSE
28 C
29 X3=XE
30 ATERM1=0.0
31 ATERM2=0.0
32 ATERM3=0.0
33 C
34 DO 160 ICHAR=1,2
35 DO 160 M=1,MMAX
36 CALL MAP (2,LMAX,M,AL,BE,DE,L1,AL1,BE1,DE1)
37 U1=U(LMAX,M,N1)
38 U2=U1
39 A1=SQRT(GAMMA*P(LMAX,M,N1)/RO(LMAX,M,N1))
40 A2=A1
41 IF (ICHAR.EQ.2) GO TO 10
42 U(LMAX,M,N3)=U1
43 A3=A1
44 C
45 C CALCULATE THE PROPERTY INTERPOLATING POLYNOMIAL COEFFICIENTS
46 C
47 10 BU=(U(LMAX,M,N1)-U(L1,M,N1))*DXR
48 BV=(V(LMAX,M,N1)-V(L1,M,N1))*DXR
49 BP=(P(LMAX,M,N1)-P(L1,M,N1))*DXR
50 BRO=(RO(LMAX,M,N1)-RO(L1,M,N1))*DXR
51 BYCB=(YCB(LMAX)-YCB(L1))*DXR
52 BAL=(AL-AL1)*DXR
53 BBE=(BE-BE1)*DXR
54 BDE=(DE-DE1)*DXR
55 CU=U(LMAX,M,N1)-BU*X3
56 CV=V(LMAX,M,N1)-BV*X3
57 CP=P(LMAX,M,N1)-BP*X3
58 CRO=RO(LMAX,M,N1)-BRO*X3
59 CYCB=YCB(LMAX)-BYCB*X3
60 CAL=AL-BAL*X3
61 CBE=BE-BBE*X3
62 CDE=DE-BDE*X3
63 C
64 C CALCULATE THE CROSS DERIVATIVE INTERPOLATING POLYNOMIAL
65 C COEFFICIENTS
66 C
67 IF (M.EQ.1) GO TO 20
68 DU=(U(LMAX,M,N1)-U(LMAX,M-1,N1))*DYR
69 DV=(V(LMAX,M,N1)-V(LMAX,M-1,N1))*DYR
70 DP=(P(LMAX,M,N1)-P(LMAX,M-1,N1))*DYR
71 DRO=(RO(LMAX,M,N1)-RO(LMAX,M-1,N1))*DYR
72 DU1=(U(L1,M,N1)-U(L1,M-1,N1))*DYR
73 DV1=(V(L1,M,N1)-V(L1,M-1,N1))*DYR
74 DP1=(P(L1,M,N1)-P(L1,M-1,N1))*DYR
75 DRO1=(RO(L1,M,N1)-RO(L1,M-1,N1))*DYR
76 GO TO 80
77 20 IF (LECB.EQ.LMAX) GO TO 30
78 DU=0.0
79 DV=V(LMAX,2,N1)*DYR
80 DP=0.0
81 DRO=0.0
82 DU1=0.0
83 DV1=V(L1,2,N1)*DYR
84 DP1=0.0
85 DRO1=0.0
86 GO TO 80
87 30 DU=(U(LMAX,2,N1)-U(LMAX,1,N1))*DYR
88 DV=(V(LMAX,2,N1)-V(LMAX,1,N1))*DYR

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89      DP=(P(LMAX,2,N1)-P(LMAX,1,N1))*DYN
90      DRO=(RO(LMAX,2,N1)-RO(LMAX,1,N1))*DYN
91      DU1=(U(L1,2,N1)-U(L1,1,N1))*DYN
92      DV1=(V(L1,2,N1)-V(L1,1,N1))*DYN
93      DP1=(P(L1,2,N1)-P(L1,1,N1))*DYN
94      DRO1=(RO(L1,2,N1)-RO(L1,1,N1))*DYN
95  80    BDU=(DU-DU1)*DXR
96      BDV=(DV-DV1)*DXR
97      BDP=(DP-DP1)*DXR
98      BDRO=(DRO-DRO1)*DXR
99      CDU=DU-BDU*X3
100     CDV=DV-BDV*X3
101     CDP=DP-BDP*X3
102     CDRO=DRO-BDRO*X3
103  C
104  C   CALCULATE X1 AND X2
105  C
106     IF (ICHR.EQ.2) A3=SQRT(GAMMA*P(LMAX,M,N3)/RO(LMAX,M,N3))
107     DO 50 IL=1,2
108     X1=X3-(U(LMAX,M,N3)-U1)*0.5*DT
109     X2=X3-(U(LMAX,M,N3)-A3+U2-A2)*0.5*DT
110  C
111  C   INTERPOLATE FOR THE PROPERTIES
112  C
113     U1=BU*X1+CU
114     U2=BU*X2+CU
115     P2=BP*X2+CP
116     RO2=BRO*X2+CRO
117     A2=SQRT(GAMMA*P2/RO2)
118  50    CONTINUE
119     V1=BV*X1+CV
120     P1=BP*X1+CP
121     RO1=BRO*X1+CRO
122     YCB1=BYCB*X1+CYCB
123     AL1=BAL*X1+CAL
124     BE1=BBE*X1+CBE
125     DE1=BDE*X1+CDE
126     UV1=U1*AL1+V1*BE1+DE1
127     A1=SQRT(GAMMA*P1/RO1)
128     V2=BV*X2+CV
129     YCB2=BYCB*X2+CYCB
130     AL2=BAL*X2+CAL
131     BE2=BBE*X2+CBE
132     DE2=BDE*X2+CDE
133     UV2=U2*AL2+V2*BE2+DE2
134  C
135  C   INTERPOLATE FOR THE CROSS DERIVATIVES
136  C
137     DU1=BDU*X1+CDU
138     DV1=BDV*X1+CDV
139     DP1=BDP*X1+CDP
140     DRO1=BDRO*X1+CDRO
141     DU2=BDU*X2+CDU
142     DV2=BDV*X2+CDV
143     DP2=BDP*X2+CDP
144     DRO2=BDRO*X2+CDRO
145  C
146  C   CALCULATE THE PSI TERMS
147  C
148     IF (NDIM.EQ.0) GO TO 70
149     IF (M.EQ.1.AND.LECB.NE.IMAX) GO TO 60
150     ATERM1=RO1*V1/((DY*(M-1))/BE1+YCB1)
151     ATERM2=RO2*V2/((DY*(M-1))/BE2+YCB2)
152     GO TO 70
153  60    ATERM1=RO1*BE1*DV1
154     ATERM2=RO2*BE2*DV2
155  70    PSI31=-UV1*DV1-BE1*DP1/RO1
156     PSI41=-UV1*DP1+A1*A1*UV1*DRO1
157     PSI12=-UV2*DRO2-RO2*AL2*DU2-RO2*BE2*DV2-ATERM2
158     PSI22=-UV2*DU2-AL2*DP2/RO2
159     PSI42=-UV2*DP2+A2*A2*UV2*DRO2
160     IF (ICHR.EQ.1) GO TO 130
161  C
162  C   CALCULATE THE CROSS DERIVATIVES AT THE SOLUTION POINT
163  C
164     IF (M.EQ.1.AND.LECB.NE.IMAX) GO TO 88

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```

165      IF (M.EQ.MMAX) GO TO 98
166      DU3=(U(LMAX,M+1,N3)-U(LMAX,M,N3))*DYS
167      DV3=(V(LMAX,M+1,N3)-V(LMAX,M,N3))*DYS
168      DP3=(P(LMAX,M+1,N3)-P(LMAX,M,N3))*DYS
169      DRO3=(RO(LMAX,M+1,N3)-RO(LMAX,M,N3))*DYS
170      GO TO 100
171 88      DU3=0.0
172      DV3=V(LMAX,2,N3)*DYS
173      DP3=0.0
174      DRO3=0.0
175      GO TO 100
176 98      DU3=(U(LMAX,MMAX,N3)-U(LMAX,M1,N3))*DYS
177      DV3=(V(LMAX,MMAX,N3)-V(LMAX,M1,N3))*DYS
178      DP3=(P(LMAX,MMAX,N3)-P(LMAX,M1,N3))*DYS
179      DRO3=(RO(LMAX,MMAX,N3)-RO(LMAX,M1,N3))*DYS
180 C
181 C      CALCULATE THE PSI TERMS AT THE SOLUTION POINT
182 C
183 100      IF (NDIM.EQ.0) GO TO 120
184      IF (M.EQ.1.AND.LECB.NE.LMAX) GO TO 110
185      ATERM3=RO(LMAX,M,N3)*V(LMAX,M,N3)/((DY*(M-1))/BE+YCB(LMAX))
186      GO TO 120
187 110      ATERM3=RO(LMAX,1,N3)*BE*DV3
188 120      UV3=U(LMAX,M,N3)*AL+V(LMAX,M,N3)*BE+DE
189      PSI13=-UV3*DRO3-RO(LMAX,M,N3)*(AL*DU3+BE*DV3)-ATERM3
190      PSI23=-UV3*DU3-AL*DP3/RO(LMAX,M,N3)
191      PSI33=-UV3*DV3-BE*DP3/RO(LMAX,M,N3)
192      PSI43=-UV3*DP3+A3*A3*UV3*DRO3
193 C
194 C      CALCULATE THE COMPATIBILITY EQUATION COEFFICIENTS
195 C
196 130      IF (ICHR.EQ.1) GO TO 140
197      PSI31B=(PSI31+PSI33)*0.5
198      PSI41B=(PSI41+PSI43)*0.5
199      PSI12B=(PSI12+PSI13)*0.5
200      PSI22B=(PSI22+PSI23)*0.5
201      PSI42B=(PSI42+PSI43)*0.5
202      GO TO 150
203 140      PSI31B=PSI31
204      PSI41B=PSI41
205      PSI12B=PSI12
206      PSI22B=PSI22
207      PSI42B=PSI42
208 C
209 C      SOLVE THE COMPATIBILITY EQUATIONS FOR U, V, P, AND RO
210 C
211 150      P(LMAX,M,N3)=PE
212      RO(LMAX,M,N3)=RO1+(P(LMAX,M,N3)-P1-DT*PSI41B)/(A3*A3+A1*A1)
213      U(LMAX,M,N3)=U2+(((PSI42B+(RO2-RO(LMAX,M,N3))*(A2+A3)*PSI22B/4.0
214      1 -(A2*A2+A3*A3)*PSI12B/2.0)*DT-(P(LMAX,M,N3)-P2))
215      2 /(RO2+RO(LMAX,M,N3))/(A2+A3))*4.0
216      V(LMAX,M,N3)=V1+DT*PSI31B
217 160      CONTINUE
218 C
219      V(LMAX,MMAX,N3)=-U(LMAX,MMAX,N3)*NXNY(LMAX)
220      V(LMAX,1,N3)=-U(LMAX,1,N3)*NXNYCB(LMAX)
221      IF (JFLAG.EQ.0) RETURN
222      V(LMAX,MMAX,N3)=V(LMAX,MMAX,N3)+XWI(LMAX)
223      RETURN
224      END

```

B.14 Visualization Output (plot.f)

This subroutine generates visualization output including velocity vector plots and contour plots of dependent variables (pressure, density, entropy). It creates plotter data for external visualization tools. *OCR note: this listing was reconstructed from Appendix C PDF text blocks for readability; contour-cell intersection logic was restored from degraded scan lines with normalized fixed-form token syntax.*

```

1  SUBROUTINE PLOT(TITLE,T,NP)
2  C
3  C *****
4  C THIS SUBROUTINE PLOTS THE VELOCITY VECTORS AND DEPENDENT
5  C VARIABLE CONTOUR PLOTS
6  C *****
7  C
8  C OCR note:
9  C Reconstructed from Appendix C PDF text blocks. Contour-cell
10 C intersection logic (labels 240-320) was restored from degraded
11 C scan blocks with normalized fixed-form token syntax.
12 C
13 DIMENSION CQ(81,21),CON(9),XCO(4),YCO(4),TITLE(8)
14 COMMON /AV/ IAV,CAV,NST,SMP,LSS,CTA,XMU,XLA,RKMU,QUT(81,21),QVT(81,21),QPT(81,21)
15 COMMON /ONESID/ UD(4),VD(4),PD(4),ROD(4)
16 COMMON /SOLUTN/ U(81,21,2),V(81,21,2),P(81,21,2),RO(81,21,2)
17 COMMON /CNTRL/ LMAX,MMAX,NMAX,NPRINT,TCONV,FDT,GAMMARGAS,GAMI,GAM2,
18 1L1,L2,L3,M1,M2,DX,DY,DT,N,N1,N3,NASM,IVEL,ICHAR,N1D,LJET,JFLAG,
19 2IERR,IUI,IUO,DXR,DYR,LD,MD,LMD1,LMD3,IB,RSTAR,RSTARS,NPLOT,G,PC,TC,
20 3LC,FLOW,ROLOW
21 COMMON /GEMIRYC/ NGEOM,XI,RI,XT,RT,XE,RE,RCI,RCT,ANGI,ANGE,XW(81),
22 1YW(81),XWI(81),YWI(81),NXNY(81),NWPTS,IINT,IDIF,LT,NDIM
23 COMMON /GCB/ NGCB,XICB,RICB,XTCB,RTCB,XECB,RECB,RCICB,RCTCB,ANGICB,
24 1ANGEGB,XCB(81),YCB(81),XCBI(81),YCBI(81),NXNYCB(81),NCBPTS,IINTCB,
25 2IDIFCB,LECB
26 COMMON /BOC/ PT(21),TT(21),THETA(21),PE,MASSE,MASSI,MASST,THRUST,
27 1NSTAG
28 REAL MN3,NXNY,MASSI,MASST,NXNYCB,MASSE
29 C
30 C GENERATE THE VELOCITY VECTOR PLOT
31 C
32 ND=N3
33 IF (N.EQ.0) ND=1
34 CALL GETG (4HKJBN,JNM)
35 XL=XI
36 XR=XE
37 YT=YW(1)
38 YB=YCB(1)
39 DO 10 L=2,LMAX
40 YT=AMAX1(YT,YW(L))
41 YB=AMIN1(YB,YCB(L))
42 10 CONTINUE
43
44 VV=0.9*DX
45 FIYB=916.0
46 XD=(XR-XL)/(YT-YB)
47 FIR=(1022.0-1022.0/FLOAT(L1)-1.0)/900.0
48 IF (XD.LE.FIR) GO TO 20
49 FIXL=0.0
50 FIXR=1022.0-1022.0/FLOAT(L1)-1.0
51 FIYT=916.0-FIXR/XD
52 GO TO 30
53 20 FIXL=511.0-450.0*XD
54 FIXR=511.0+450.0*XD
55 FIYT=16.0
56 30 XCONV=(FIXR-FIXL)/(XR-XL)
57 YCONV=(FIYT-FIYB)/(YT-YB)
58
59 VMAX=0.0
60 DO 40 L=1,LMAX
61 DO 40 M=1,MMAX
62 VMAX=AMAX1(VMAX,ABS(U(L,M,ND)),ABS(V(L,M,ND)))
63 40 CONTINUE
64 IF (VMAX.LT.1.0E-10) GO TO 60
65
66 DROU=VV/VMAX
67 CALL ADV (1)
68 DO 50 L=1,LMAX
69 IX1=FIXL+(FLOAT(L-1)*DX)*XCONV
70 DYL=(YW(L)-YCB(L))/FLOAT(MMAX+1)
71 DO 50 M=1,MMAX
72 IY1=FIYB+(YCB(L)+FLOAT(M-1)*DYL-YB)*YCONV
73 IX2=FIXL+(FLOAT(L-1)*DX+U(L,M,ND)*DROU)*XCONV
74 IY2=FIYB+(YCB(L)+FLOAT(M-1)*DYL-YB+V(L,M,ND)*DROU)*YCONV
75 CALL DRV (IX1,IY1,IX2,IY2)

```

```

76      CALL PLT (IX1,IY1,16)
77 50    CONTINUE
78      CALL LINCNT (59)
79      WRITE (7,410)
80      WRITE (7,550) JNM,TITLE,NP,T
81 C
82 C      GENERATE THE CONTOUR PLOTS
83 C
84 60      I=0
85 70      I=I+1
86      GO TO (80,100,120,140,340), I
87
88 80      DO 90 L=1,IMAX
89          DO 90 M=1,MMAX
90              CQ(L,M)=RO(L,M,ND)*G
91 90      CONTINUE
92      GO TO 160
93
94 100     DO 110 L=1,IMAX
95         DO 110 M=1,MMAX
96             CQ(L,M)=P(L,M,ND)/PC
97 110     CONTINUE
98         GO TO 160
99
100 120     DO 130 L=1,IMAX
101         DO 130 M=1,MMAX
102             CQ(L,M)=P(L,M,ND)/RO(L,M,ND)/RGAS/G-TC
103 130     CONTINUE
104         GO TO 160
105
106 140     DO 150 L=1,IMAX
107         DO 150 M=1,MMAX
108             CQ(L,M)=SQRT(U(L,M,ND)**2+V(L,M,ND)**2)/(GAMMA*P(L,M,ND)/RO(L,M,ND))
109 150     CONTINUE
110
111 160     QMN=1.0E06
112     QMX=-QMN
113     DO 170 L=1,IMAX
114         DO 170 M=1,MMAX
115             QMN=AMIN1(CQ(L,M),QMN)
116             QMX=AMAX1(CQ(L,M),QMX)
117 170     CONTINUE
118     XX=QMX*QMN
119     DQ=0.1*XX
120     DO 180 K=1,9
121         CON(K)=QMN*FLOAT(K)*DQ
122 180     CONTINUE
123         K=9
124
125     CALL ADV (1)
126     CALL LINCNT (59)
127     GO TO (190,200,210,220), I
128 190     WRITE (7,560)
129     GO TO 230
130 200     WRITE (7,570)
131     GO TO 230
132 210     WRITE (7,580)
133     GO TO 230
134 220     WRITE (7,590)
135 230     WRITE (7,400) QMN,QMX,CON(1),CON(K),DQ
136     WRITE (7,550) JNM,TITLE,NP,T
137 C
138     DO 520 L=2,IMAX
139         DY=(YW(L-1)-YCB(L-1))/FLOAT(MMAX-1)
140         DY1=(YW(L)-YCB(L))/FLOAT(MMAX-1)
141     DO 520 M=2,MMAX
142         NN=0
143         DO 520 KK=1,K
144             K1=0
145             K2=0
146             K3=0
147             K4=0
148             IF (CQ(L-1,M-1).LE.CON(KK)) K1=1
149             IF (CQ(L,M-1).LE.CON(KK)) K2=1
150             IF (CQ(L-1,M).LE.CON(KK)) K3=1
151             IF (CQ(L,M).LE.CON(KK)) K4=1

```

```

152      IF (K1*K2*K3*K4.NE.0) GO TO 520
153      IF (K1+K2+K3+K4.EQ.0) GO TO 520
154      IF (NN.NE.0) GO TO 240
155      NN=1
156      XCO(1)=X1+FLOAT(L-2)*DX
157      XCO(2)=XCO(1)+DX
158      XCO(3)=XCO(1)
159      XCO(4)=XCO(2)
160      YCO(1)=YCB(L-1)+FLOAT(M-2)*DY
161      YCO(2)=YCB(L)+FLOAT(M-2)*DY1
162      YCO(3)=YCB(L-1)+FLOAT(M-1)*DY
163      YCO(4)=YCB(L)+FLOAT(M-1)*DY1
164 240    LL=0
165      IF (K1+K3.NE.1) GO TO 250
166      IC1=1
167      IC2=3
168      LP1=L-1
169      MP1=M-1
170      LP2=L-1
171      MP2=M
172      ASSIGN 250 TO KR1
173      GO TO 280
174 250    IF (K1+K2.NE.1) GO TO 260
175      IC1=1
176      IC2=2
177      LP1=L-1
178      MP1=M-1
179      LP2=L
180      MP2=M-1
181      ASSIGN 260 TO KR1
182      GO TO 280
183 260    IF (K2+K4.NE.1) GO TO 270
184      IC1=2
185      IC2=4
186      LP1=L
187      MP1=M-1
188      LP2=L
189      MP2=M
190      ASSIGN 270 TO KR1
191      GO TO 280
192 270    IF (K3+K4.NE.1) GO TO 520
193      IC1=3
194      IC2=4
195      LP1=L-1
196      MP1=M
197      LP2=L
198      MP2=M
199      ASSIGN 520 TO KR1
200 280    LL=LL+1
201      XX=(CON(KK)-CQ(LP1,MP1))/(CQ(LP2,MP2)-CQ(LP1,MP1))
202      IF (LL.EQ.2) GO TO 290
203      IX1=FIXL+(XCO(IC1)+XX*(XCO(IC2)-XCO(IC1))-XL)*XCONV
204      IY1=FIYB+(YCO(IC1)+XX*(YCO(IC2)-YCO(IC1))-YB)*YCONV
205      GO TO KR1,(250,260,270,320)
206 290    IX2=FIXL+(XCO(IC1)+XX*(XCO(IC2)-XCO(IC1))-XL)*XCONV
207      IY2=FIYB+(YCO(IC1)+XX*(YCO(IC2)-YCO(IC1))-YB)*YCONV
208      CALL DRV (IX1,IY1,IX2,IY2)
209      IF (KK.NE.1) GO TO 300
210      CALL PLT (IX1,IY1,35)
211 300    IF (KK.NE.K) GO TO 310
212      CALL PLT (IX1,IY1,24)
213 310    LL=0
214      IF (LP2.NE.L) GO TO 520
215      IF (MP2.NE.M) GO TO 320
216      GO TO 260
217 320    CONTINUE
218      CONTINUE
219 520    CONTINUE
220
221      DO 330 L=2,LMAX
222      IX1=FIXL+(FLOAT(L-2)*DX)*XCONV
223      IX2=FIXL+(FLOAT(L-1)*DX)*XCONV
224      IY1=FIYB+(YCB(L-1)-YB)*YCONV
225      IY2=FIYB+(YCB(L)-YB)*YCONV
226      IY3=FIYB+(YW(L-1)-YB)*YCONV
227      IY4=FIYB+(YW(L)-YB)*YCONV

```

```

228      CALL DRV (IX1,IY1,IX2,IY2)
229      CALL DRV (IX1,IY3,IX2,IY4)
230 330   CONTINUE
231      GO TO 70
232
233 340   DY=1.0/FLOAT(MMAX*1)
234      CALL ADV (1)
235      RETURN
236
237 400   FORMAT (1H ,10HLOW VALUE=,1PE11.4,2X,11HHIGH VALUE=,E11.4,2X,
238          1 12HLOW CONTOUR=,E11.4,2X,13HHIGH CONTOUR=,E11.4,2X,
239          2 14HDELTA CONTOUR=,E11.4)
240 410   FORMAT (1H ,16HVELOCITY VECTORS)
241 550   FORMAT (1H ,A10,4X,8A10,2X,2HN=,I4,2X,2HT=,1PE10.4,4H SEC)
242 560   FORMAT (1H ,7HDENSITY)
243 570   FORMAT (1H ,8HPRESSURE)
244 580   FORMAT (1H ,11HTEMPERATURE)
245 590   FORMAT (1H ,11HMACH NUMBER)
246      END

```


Appendix A

Characteristic Relations: η Constant Plane

Introduction

This appendix derives the characteristic relations for the $\eta = \text{constant}$ reference plane. These relations are used to implement inlet boundary conditions in the NAP solver.

I. Equations of Motion

The equations of motion can be written in the form:

$$\frac{\partial P}{\partial t} + u \frac{\partial P}{\partial \xi} + v \frac{\partial P}{\partial \eta} = -vP_\eta - pau_\xi - pBv_\xi - \frac{epvB}{n} \quad (\text{A.1})$$

$$\frac{\partial u}{\partial t} + u \frac{\partial u}{\partial \xi} + \frac{P_\xi}{p} = -vu_\eta - ap_\xi/p \quad (\text{A.2})$$

$$\frac{\partial v}{\partial t} + uv_\xi + \frac{BP_\eta}{p} = -vv_\eta - \frac{eP_\eta}{p} \quad (\text{A.3})$$

$$\frac{\partial P}{\partial t} + up_\xi - a^2 \left(\frac{\partial P}{\partial t} + u \frac{\partial P}{\partial \xi} \right) = -vP_\eta + a^2 vp_\eta \quad (\text{A.4})$$

II. Characteristic Curves

The characteristic curves are derived from analysis of the system's hyperbolicity:

$$\frac{d\eta}{dT} = v \quad (\text{A.5})$$

$$\frac{d\eta}{dx} = \frac{\partial \eta}{\partial x} \quad (\text{A.6})$$

III. Characteristic Variables

Define the characteristic variables:

$$\psi_1 = vP_\xi - \rho au_\xi - pBv_\xi - \frac{epvB}{n} \quad (\text{A.7})$$

$$\psi_2 = -vu_\xi - aP_\xi/p \quad (\text{A.8})$$

$$\psi_3 = -vv_\xi - \frac{eP_\xi}{p} \quad (\text{A.9})$$

$$\psi_4 = -vP_\xi + a^2vP_\xi \quad (\text{A.10})$$

IV. Compatibility Equations

Substituting the characteristic equations yields compatibility relations. For the characteristic curve with slope $d\eta = (u - a)dT$:

$$dp - \rho adu = (\psi_2 + 2\psi_1 - \rho a\psi_2)dT \quad (\text{A.11})$$

For the characteristic curve with slope $d\eta = (u + a)dT$:

$$dp + \rho adu = (\psi_4 + 2\psi_1 + \rho a\psi_2)dT \quad (\text{A.12})$$

These relations provide the basis for implementing inlet boundary conditions through the method of characteristics.

Note: Technical content in this appendix was extracted via OCR from the original NAP document. Equation symbols, indices, and coordinate transformations have been verified against the method description in Chapter I but should be confirmed against the original source for critical applications.

Appendix B

Characteristic Relations: ζ Constant Plane

Introduction

This appendix derives the characteristic relations for the $\zeta = \text{constant}$ reference plane. These relations are used to implement wall and centerbody boundary conditions in the NAP solver.

I. Equations of Motion

The equations of motion for the ζ -plane are:

$$\frac{\partial P}{\partial t} + v \frac{\partial P}{\partial \eta} + \rho a u_\eta + \rho B v_\eta = -u P_\xi - \rho u_\xi - \frac{epvB}{n} \quad (\text{B.1})$$

$$\frac{\partial u}{\partial t} + v u_\eta + \frac{a P_\eta}{\rho} = -u u_\xi - \frac{P_\xi}{\rho} \quad (\text{B.2})$$

$$\frac{\partial v}{\partial t} + v v_\eta + \frac{B P_\eta}{\rho} = -u v_\xi \quad (\text{B.3})$$

$$\frac{\partial P}{\partial t} + v P_\eta - a^2 \left(\frac{\partial P}{\partial t} + v \frac{\partial P}{\partial \eta} \right) = -u P_\xi + a^2 u P_\xi \quad (\text{B.4})$$

II. Characteristic Curves

Following the development of Appendix A, the characteristic curves for the ζ -plane are:

$$\frac{d\zeta}{dT} = v \quad (\text{B.5})$$

$$\frac{d\zeta}{dx} = v \pm a^* a \quad (\text{B.6})$$

where $a^* = (a^2 + B^2)^{1/2}$ represents the effective sound speed in the transformed coordinate system.

III. Compatibility Equations

The compatibility equations for the ζ -constant plane are:

$$adu - \rho dv = (\psi_0 - a\psi_1)dT \quad (\text{B.7})$$

$$dp - a^2 d\rho = \psi_4 d\xi \quad (\text{B.8})$$

$$dp - \rho a^2 \frac{du}{a^*} - \rho B a \frac{dv}{a^*} = \left(\psi_2 + a\psi_1 - \frac{\rho a a \psi_0}{a^*} - \frac{\rho B a \psi_1}{a^*} \right) dT \quad (\text{B.9})$$

$$dp + \rho a^2 \frac{du}{a^*} + \rho B a \frac{dv}{a^*} = \left(\psi_3 + a\psi_1 + \frac{\rho a a \psi_0}{a^*} + \frac{\rho B a \psi_1}{a^*} \right) dT \quad (\text{B.10})$$

These compatibility equations apply along the characteristic curves and provide the boundary condition implementation for wall and centerbody surfaces.

Note: Technical content in this appendix was extracted via OCR from the original NAP document and reconstructed using the methods described in Chapter I.E. Coordinate transformations and all equations should be verified against reference material before use in alternative implementations.