

Spacecraft dynamics project to design a zero
momentum biased satellite.

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Part I

Problem Statement

Design a preliminary attitude control system for a satellite. The satellite can have any of the following stabilisation methods given below.

1. Gravity gradient stabilisation
2. Spin/dual spin stabilisation with passive/active magnetic torque for damping.
3. Momentum biased stabilisers with earth sensors measuring roll and pitch as primary sensors with gyroscopes and schemes for momentum dumping using thrusters.
4. 0 momentum biased spacecraft with star sensor for roll, pitch and yaw euler parameters with gyroscopes along with momentum dumping mechanism of wheels.

Steps involved:

1. Select a suitable kinematic system for spacecraft.
2. Using Euler's equation derive dynamical equations of motion and include gravity gradient torque.
3. Study stability dynamics of the system of both pitch motion and roll- yaw motion and figure out what kind of motion is possible. Also describe why active control system is required.
4. Design a control system accordingly to control spacecraft with PID strategies.
5. Figure out a control strategy to momentum dump with selected wheel based control.
6. Select proportional control gain accounting for maximum allowable steady state of 0.005 deg about all axes (for zero-momentum biased system).

Problem Assigned:

Oceansat-1 is a 3-axis stabilized earth pointing satellite with a 4-wheel configuration which is traditional. That is the wheel configuration is with a wheel about each principal axis and the 4th wheel is mounted with 54.7 deg with respect to all three wheels. Nominally the principal axis wheels are rotated with 1000 rpm and the redundant wheel is rotated with -1732 rpm so that zero-momentum is achieved.

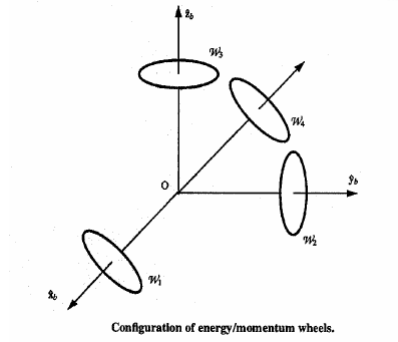


Figure 1: Configuration of wheels

The momentum dumping is achieved by using 60 Am^2 torque rods about all the three axes. Actual MI properties of the s/c after deployment are,

$$J_c = \begin{bmatrix} 1800 & -50 & -15 \\ -50 & 1600 & 25 \\ -15 & 25 & 1200 \end{bmatrix} \text{ Kg m}^2 \quad (1)$$

Where the mass is given to be 1600 Kg, and $[x,y,z]$ correspond to yaw, roll and pitch axes respectively.

Initially assume that the cross product of inertias is negligible and design the control system. Then when you actually apply the control, use the actual inertia matrix and compare and comment how the performance varies.

Use momentum dumping by torque rods about 2 axes and design PID control for $T_x = T_z = 2 * 10^{-3} \text{ Nm}$ and $T_y = 10^{-4} \text{ Nm}$ with $\omega_0 = 1.0741 * 10^{-3} \text{ rad/s}$. Also compare strategy and time responses for tetrahedron and Pyramid configurations.



Figure 2: Ocean Sat

Part II

Kinematics of spacecraft

The kinematics for the spacecraft is modelled using Euler angles since the spacecraft is expected to operate within small ranges of Euler angles and it is easier to model the desired orientation using the Euler angles convention.

The Dynamical Equations of motion for the spacecraft are given as follows.

$$I\dot{\omega}_b + \left(\omega_b - C_{b0} \begin{bmatrix} 0 \\ 0 \\ \omega_0 \end{bmatrix} \right) \times I \left(\omega_b - C_{b0} \begin{bmatrix} 0 \\ 0 \\ \omega_0 \end{bmatrix} \right) = T_{disturbance} + T_{magnetic} - T_{RW} \quad (2)$$

Where T represents Torque and $T_{magnetic}$ represents torque by magnetic torquers due to momentum dumping and T_{RW} represents torque due to reaction wheels.

For reaction wheels, the net torque due to the wheels come out as,

$$T_{RW} = I_w \begin{bmatrix} \dot{\omega}_1 - \frac{\omega_4}{\sqrt{3}} \\ \dot{\omega}_2 - \frac{\omega_4}{\sqrt{3}} \\ \dot{\omega}_3 - \frac{\omega_4}{\sqrt{3}} \end{bmatrix} + I_w \begin{bmatrix} \omega_3\omega_{by} - \omega_2\omega_{bz} \\ \omega_1\omega_{bz} - \omega_3\omega_{bx} \\ \omega_2\omega_{bx} - \omega_1\omega_{by} \end{bmatrix} \quad (3)$$

Where, ω_b is given by,

$$\omega_b = \begin{bmatrix} -\sin(r) & 0 & 1 \\ \cos(r)\sin(y) & \cos(y) & 0 \\ \cos(r)\cos(y) & -\sin(y) & 0 \end{bmatrix} \begin{bmatrix} \dot{p} \\ \dot{r} \\ \dot{y} \end{bmatrix}$$

where $[\mathbf{x}, \mathbf{y}, \mathbf{z}]$ correspond to the yaw(y), roll(r) and pitch(p) axes respectively and C_{b0} corresponds to the rotation matrix by,

$$C_{b0} = C_x(p)C_y(r)C_z(y) \quad (4)$$

by the 3-2-1 rotation convention.

The gravity gradient torque is given by,

$$T_g = \frac{3\mu}{r^5} r_b^\times I r_b \quad (5)$$

This is accounted for in the disturbance torque.

The magnetic torque is based on a magnetic dipole system model of the earth where the satellite is in orbit about. Hence, for any point in the orbit, the magnetic field around the satellite can be easily found using the model and assuming that the dipole is aligned at angle of 11° wrt earth's true axis, we find the radial and azimuthal fields to be,

$$B_r = -2B_0 \left(\frac{R_E}{r} \right)^3 \cos(\theta) \quad (6)$$

$$B_\theta = -B_0 \left(\frac{R_E}{r} \right)^3 \sin(\theta) \quad (7)$$

$$|B| = B_0 \left(\frac{R_E}{r} \right)^3 \sqrt{1 + 3\cos^2(\theta)} \quad (8)$$

where, R_E is the mean radius of the earth, θ is the azimuth from the north magnetic pole, r is the distance from earth's center and $B_0 = 3.12 \times 10^{-5} T$. There is no need to consider solar wind effects since the satellite OCEANSAT revolves closely about earth.