



ESEIAAT



Escola Superior d'Enginyeries Industrials,  
Aeroespacial i Audiovisual de Terrassa

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# Cubesat Constellation Astrea

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## Report

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### Students:

Cebrián Galán, Joan

Foreman Campins, Lluís

Fuentes Muñoz, Óscar

Herrán Albelda, Fernando

Martínez Viol, Víctor

Pla Olea, Laura

Puig Ruiz, Josep

Tarroç Gil, Sergi

Urbano González, Eva María

Fontanes Molina, Pol

Fraixedas Lucea, Roger

González García, Sílvia

Kaloyanov Naydenov, Boyan

Morata Carranza, David

Pons Daza, Marina

Serra Moncunill, Josep Maria

Tió Malo, Xavier

**Customer:** Pérez Llera, Luís Manuel



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## Part I

# Satellite design

# Chapter 1

## Satellite design

### 1.1 Structure and mechanics

The design and operation of a CubeSat is a complex process that must be completed keeping in mind the huge differences between all subsystems as well as the role they will play during the lifetime of the mission. And since these systems will operate in space, they have to be prepared and certified to withstand extreme temperature and radiation conditions.

The satellite used by Astrea must have high compatibility between all the systems to avoid potential problems and has to be tested (either all the systems together or one by one). Their correct functioning has to be ensured, especially the critical systems such as the solar arrays, batteries and antennas should be fully operational for at least four years.

#### 1.1.1 Structure

The mission of the structure is to sustain and protect all the electronic devices carried by the satellite. In order to ensure that all the electronic and mechanical systems can be mounted upon the structure, a high compatibility between these systems is required; therefore, the structure must be very flexible regarding the arrangement of the subsystems.

The structure chosen is manufactured by **Innovative Solutions In Space (ISIS)**. Among its features it is worth mentioning that it can withstand the high range of temperature it will face in the space (from  $-40^{\circ}\text{C}$  to  $80^{\circ}\text{C}$ ) and it is highly compatible; almost every physical system used can be placed within the structure or on its faces (such

as the antennas or the deployable solar arrays). Finally, the mass of the structure is relatively low, and given that the mass of the other subsystems is sometimes a drawback, it is plus point.

### 1.1.2 Thermal protection

The CubeSat is vulnerable to suffer extreme temperatures while operating in space, both below zero and above zero. The thermal protection system consists of a set of layers (MLI) made of insulating materials and it aims to protect the CubeSat from potential thermal shocks. The satellite must remain within an optimal range of temperature, despite of the variation of the external temperature, in order to work properly. Furthermore, the thermal protection system should also dissipate the heat produced by the other systems.

**Dunmore Aerospace** has been chosen to provide us its MLI product. The product, **Dunmore Aerospace Satkit**, has been designed for small satellites operating in LEO and it will provide the CubeSat with the protection required during operation.

### 1.1.3 Options chosen for the structure and thermal protection

The options chosen are presented in the table 1.1.1.

System	Brand and model	Price per unit (€)	N. of units
3U Structure	ISIS	3900	1
Thermal Protection	Dunmore Satkit	1000	1

Table 1.1.1: Options chosen for the structure and thermal protection

## 1.2 Electrical Power System

The Electrical Power System (EPS) of the satellite must provide and manage the energy generated efficiently in order to have all the systems operating under typical conditions during the lifetime of the mission. The role of this system is to control and distribute a continuous power to the Cubesat, to protect the satellite against electrical bus failiures and to monitor and communicate the status of the EPS to the on-board computer. The EPS of the Cubesat is, probably, the most fundamental requirement of the satellite, since its failure would result in a mission failure.



### 1.2.1 Estimation of the power required

To select the adequate electrical power systems it is essential that the power consumed by the CubeSat is known *a priori*. Thus, to select the solar arrays and the batteries, as well as the power management system, an estimation of the power consumed has to be made.

The total power required is 52W and it has been estimated considering that all the subsystems are working under typical working conditions.

### 1.2.2 Solar arrays

Given that the space of a 3U CubeSat is very limited, the primary source of electrical power has to be photovoltaic cells. The photovoltaic cells will collect and convert the energy of the sun into electrical energy and they have to be fully efficient for at least four years (this is: it has to be ensured that the missions does not run out of power for at least four years).

Every cubesat will come with at least 4 deployable solar panels (manufactured by **EXA-Agencia Espacial Civil Ecuatoriana** providing it with 67.2W of power, approximately, to supply peak demands. Note that these 4 deployable solar panels are a basic requirement. If more space is available on the faces of the satellite, additional panels can be placed providing extra power.

### 1.2.3 Power management system

The role of the power management system is to distribute the power and supply the energy to the different systems used in the CubeSat. Since these systems have different power and energy needs, the power management system has to be highly compatible and must have a high enough number of buses to supply the different voltage and intensity required to the systems.

The selected option for the mission is the **NanoPower P60** by **Gomspace**, a high-power EPS for small satellites that comes with 1 motherboard, 1 ACU module (Array Conditioning Unit) and 1 PDU (Power Distribution Unit), allowing multiple configurations in just one motherboard and saving a lot of space.

### 1.2.4 Batteries

The role of the batteries is to provide the subsystems of the satellite with the power needed when the solar arrays are working less efficiently or not properly. Astrea is looking for decent capacity batteries that provide a *slightly higher* than typical energy and power supply, since all the systems will not usually operate under peak conditions.

Astrea has chosen the **BA01/D** batteries manufactured by **EXA-Agencia Espacial Civil Ecuatoriana**. The CubeSat will have two of these batteries, with a total capacity of 28800mAh or 106.4Wh.

Through the lifetime of the mission, the solar arrays will face an important unfavorable condition; in the worst case scenario, the satellite will be in the dark during half of the period of the orbit. So, it is clear that the batteries are a critical system. If the satellite was in the dark during half of the period of the orbit, the estimated energy that it would need would be 50Wh. Thereby, the capacity of the batteries is more than enough to supply the energy required in the worst case scenario. Furthermore, they will supply energy when the energy demand of the CubeSat is higher than the energy collected by the solar cells. And logically, they will store the energy collected by the solar arrays when the energy demand of the systems is lower than the energy collected.

### 1.2.5 Options chosen for the EPS

Finally, the options chosen are presented in the table 1.4.1.

System	Brand and model	Price per unit (€)	N. of units
Solar arrays	EXA	17000	4
Additional solar arrays	-	4000-12000	depends
Batteries	EXA	6300	2
Power Management	Gomspace NanoPower P60	16000	1

Table 1.2.1: Options studied for the Electric Power System

## 1.3 Propulsion Systems

The propulsion system is an important part of the satellite given the needs of the satellite to perform different maneuvers, such as reach the desired orbit after it has been deployed from the rocket, and to maintain the orbit and avoid falling to the Earth. The main

parameters that have to be consider are thrust, total specific impulse, power required, weight of the propulsion system and its volume, since the size and weight of the CubeSat are very restrictive.

At the moment, the most used and modern thrusters for small satellites are: ionic, pulsed plasma, electrothermal and green monopropellant thrusters. An important aspect to consider is that the goal is to reduce the mass required although this will imply smaller accelerations than conventional propulsion systems.

The **BGT-X5**, a green monopropellant propulsion system, has been chosen as the thruster for the CubeSat. The high thrust and delta V that BGT-X5 provides has predominated at expenses of other variables as weight or specific impulse where others options were better. With this thruster the CubeSat will be able to carry out the necessary actions to keep the satellite in orbit, to relocate the satellite or to change its orbit.

The option chosen is presented in the table below 1.3.1.

System	Brand and model	Price per unit (€)
Propulsion	Busek BGT-X5	50000

Table 1.3.1: Option chosen for the propulsion system

## 1.4 Attitude and Orbital Control Systems

The attitude and orbital control system (AOCS) is needed to enable the satellite to keep a specific position within its orbit and to control the antennas in order to remain oriented to the assigned area, because the satellite tends to change its orientation due to torque. The AOCS receives telecommands from the central computer and acquires measurements (satellite attitude and orbital position) from different sensors. The AOCS can also be referred as the ADACS (Attitude Determination and Attitude control system).

Attitude control for CubeSats relies on miniaturizing technology without significant performance degradation. Tumbling typically occurs as soon as a CubeSat is deployed, due to asymmetric deployment forces and bumping with other CubeSats. Some CubeSats operate normally while tumbling, but those that require pointing to a certain direction or cannot operate safely while spinning, must be detumbled. Systems that perform attitude determination and control include **reaction wheels**, **magnetorquers**, **thrusters**, **star trackers**, **Sun sensors**, **Earth sensors**, **angular rate sensors**, and **GPS receivers**

**and antennas.** Combinations of these systems are typically seen in order to take each method's advantages and mitigate their shortcomings. (*wikipedia extract*, [?]).

Pointing to a specific direction is necessary for Earth observation, orbital maneuvers, maximizing solar power, and some scientific instruments. Directional pointing accuracy can be achieved by sensing Earth and its horizon, the Sun, or specific stars. Determination of a CubeSat's location can be done through the use of on-board GPS, which is relatively expensive for a CubeSat, or by relaying radar tracking data to the craft from Earth-based tracking systems (*wikipedia extract*, [?]).

### 1.4.1 Orbital Control

Orbital control will be achieved as a combination of two systems. ADCS will orient the thrust (given by the propulsion system) and the operation will be controlled by the on-board computer. Mainly, the orbit control will be necessary to mitigate orbital debris effect on every satellite.

Taking into account several very restrictive variables: low power consumption, low, weight and size, high pointing accuracy and really versatile systems that can integrate multiple subsystems; **CUBE ADCS** is chosen. It has the lowest mass and power consumption, it also offers a higher attitude determination systems and integrates also and On-Board Computer (OBC).

Finally, the options chosen are presented in the table 1.4.1.

System	Brand and model	Price per unit (€)	N. of units
ACDS	CUBE ADCS	15000	1

Table 1.4.1: Options studied for the AOCDS

## 1.5 Payload

The payload needs to provide a radio link with the client's satellite(s) for real time data relay with no less than 25MB/s of data rate. For achieving its purpose, the payload will consist on a pack of arrays of antennas and data handling computers.

The CubeSat will have three types of radio links for transmitting in every condition the data received from the clients:

- **Space to Ground link:** Connection between satellite and Ground Station when it is possible.
- **Inter-satellite Space to Space link:** Communication between Astrea satellites for data relay, looking for the nearest satellite with Ground Station link available, to transmit the data.
- **Client Space to Space link:** Communication between client and Astrea satellites.

The radio frequencies used to establish the previous described links are regulated in [?] by frequency, bandwidth and type of communication . So, for the **Space to Ground link** frequencies from **70MHz** to **240GHz** will be used; for **Inter-satellite Space to Space link** plus data relay type of communication, frequencies are **2-2.4GHz**, **4-4,4GHz** and **22-240GHz**. Finally, **Client Space to Space link**, they exist to cases; on the one hand, the client points towards the Earth like a standard satellite, we capture its signal and make the data relay, since it is like a Space to Ground communication and also like a inter-satellite communication, we can combine the two previous restrictions. On the other hand, if the client satellite is below our constellation, we only had inter-satellite communication, therefore **Inter-satellite Space to Space link** rules are applied.

### 1.5.1 Antennas

The antennas are essential in this mission, since their role is to transmit and receive the data from other satellites as well as the ground stations. In order to provide fast and reliable communication, several options have been studied and information about their main parameters is presented below.

#### 1.5.1.1 Patch antenna

A **patch antenna** is a type of radio antenna with a low profile, which can be mounted on a flat surface, It consists of a flat rectangular sheet or "patch" of metal, mounted over a larger sheet of metal called a ground plane. They are the original type of microstrip antenna described by Howell in 1972. [?, wikipedia]

The antenna that will be equipped in the CubeSat is the patch antenna manufactured by **AntDevCov**. The satellite will come with 8 of this type of antenna; 4 of them will be

placed on each side face of the CubeSat and they will occupy a 1U face and the other 4 of them will be placed on the top and the bottom.

### 1.5.2 Payload Data Handling Systems

Every satellite will act as a router to transmit client data to the ground. This initial raw data, should be temporally stored into the satellite in order to process it, if necessary. Since, to down-link the data, first the satellites need to establish connection, data can not be directly retransmitted to other sources (Ground Station nor satellite) as it enters to the satellite. Furthermore, non loss compression algorithms can be applied to reduce the data size load and achieve higher data transmission velocities.

To sum up, Payload Data Handling System (PDHS) will be able to receive, process and send the client data, using the integrated transceivers (transmitter + receiver ) for sending the data and the PDHS computer to process it. PDHS have a hard disk associated which will temporally store the client data.

The PDHS selected for the mission is the **Nanomind Z7000** because of its two 667MHz processor can handle a high data payloads and process it at a high velocity velocities, reducing in last term delay between communications.

#### 1.5.2.1 Transceivers

A transceiver is a device comprising both a transmitter and a receiver that are combined and share common circuitry or a single housing. For the preliminary design, because we know that they should satisfy all the connectivity options, we are restricted to the S, K or higher bands for **Inter-satellite communication** and not restriction virtually at all for **Space to Ground** communication. Nevertheless, together with the communications department, X band is chosen as the frequency to talk to the floor because several factors: the use in

The **NanoCom TR-600** by **GOMspace** has been selected as the inner-satellite transceiver since its manufacturer offers it in combination with the **NanoMind Z7000**. Both integrated on a board able to hold three TR-600 transceivers and one computer. The low dimensions, high bandwidth (associated to high data rates) and low mass of TR-600 makes it a great choice for Inter-Satellite communication.

The **SWIFT-XTS** has been selected as the space-to-ground transceiver. Its high bandwidth will make possible higher communication data rates.

### 1.5.3 Options chosen for the payload module

Finally, with the aim to clarify all the information of this section, the chosen systems and components are presented in the table 1.5.1.

System	Brand and model	Price per unit (€)	N. of units
Antenna	Patch antenna AntDevCo	18000 1st (7000 others)	8
Transceiver	NanoCom TR-600	8545	3
Transceiver	SWIFT-XTS	5500	1
PDHS	NanoMind Z7000	5000	1

Table 1.5.1: Options chosen for the payload

## 1.6 Communication module

The telemetry subsystem analyses the information of the ground station and other sensors of the satellite in order to monitor the on-board conditions. With this system, the CubeSat is able to transmit the status of the on-board systems to the ground station. The command and control subsystem (TT&C) allows the ground station to control the satellite.

Every CubeSat of the constellation, will need to report its operating status to the ground and receive commands from the ground. TT&C operations will usually be performed when the satellite flights over the coverage of the constellation ground station, but since the satellites are interconnected, there is the possibility to perform this operations via data relay links between satellites. As a collaboration with the communications department, S band frequency is chosen for TT&C operations, since there is no need for high data rates, the lower band will significantly reduce the power consumption.

Communication to the ground will be perform with a NanoCom TR-600 transceiver module attached to AntDevCo Patch antenna, both configured for S band frequency communication.

## 1.7 Astrea satellite final configuration

System	Weight/unit (g)	Sizes (mm)	N. of units
<b>STRUCTURE AND MECHANICS</b>			
Structure	304.3	100 x 100 x 300	1
Thermal protection	38	Covers all	1
<b>Total</b>	<b>342.3</b>		
<b>ELECTRIC POWER SYSTEM</b>			
Solar arrays	175	98 x 83 x 8.50	4
Batteries	155	90 x 63 x 12.02	2
Power management	126	92.0 x 88.9 x 20.5	1
<b>Total</b>	<b>1136</b>		
<b>PAYLOAD</b>			
Patch antenna	30	90 x 90 x 4.35	8
Transceiver inter-satellite	16.4	65 x 40 x 6.5	3
Transceiver space to ground	101.5	86 x 86 x 45	1
Data handling system	28.3	65 x 40 x 6.5	1
Antenna Deployable	83	100 x 83 x 6.5	1
Variable	150	-	1
<b>Total</b>	<b>652</b>		
<b>AOCDS</b>			
Thruster	1350	90 x 90 x 95	1
ADACS	506	90 x 90 x 58	1
<b>Total</b>	<b>1856</b>		
<b>TOTAL ESTIMATION</b>	<b>3986.3</b>		

Additional information regarding the systems used in the CubeSat and the final configuration, calculations and decision taking can be found on the appendix.



## 1.8 Budget

System	Cost/unit (€)	Total cost (€)	N. of units
<b>Structure and mechanics</b>			
Structure	3900	3900	1
Thermal protection	1000	1000	1
<b>Total</b>		4900	
<b>Electrical power system</b>			
Solar arrays	17000	68000	4
Batteries	6300	12600	2
Power management	16000	16000	1
<b>Total</b>		96600	
<b>Payload</b>			
Patch antenna	18000 1st unit 7000 others	67000	8
Transceiver inter-satellite	8545	25635	3
Transceiver space to ground	5500	5500	1
Data handling system	5000	5000	1
Antenna Deployable	3000	3000	1
Variable expenses	4000	4000	1
<b>Total</b>		110135	
<b>AOCDS</b>			
Thruster	50000	50000	1
ADACS	15000	15000	1
<b>Total</b>		65000	
<b>Total CubeSat</b>		276635	
<b>Total estimation CubeSat</b>		297000	
<b>+Fixed cost</b>	(includes all CubeSats)	150000	

The difference between the total cost and the total estimation is due to the fact that every satellite has to go through a process to be ready for operation. This is, the CubeSat has to be assembled and has to be tested to ensure that all the systems are working properly. Thus, an estimation of the costs related with this operation has to be made.

The fixed cost for assembling the satellites will be 150000€(cost of renting the building, the electricity, ...) and an additional cost 20000€/unit, which will include the wages of the people assembling and testing the satellite and also other variable costs that may appear in the process (such as the purchase of additional solar arrays), is added to every

## Budget

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satellite. Furthermore, this extra 20000€ includes the costs of transport to launch site. The payload's variable expenses stand for cable, adaptors and several minor components that have to be purchased.

Several options have been studied for assembling and testing the satellite, and the option chosen is *OpenCosmos*. Astrea is committed to encourage the growth of the local economy and *OpenCosmos* would be a perfect partner for the mission. They provide companies and individuals with simple and affordable access to space offering integration and testing services.