



ESEIAAT



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Cubesat Constellation Astrea

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1 Satellite design planning

Interdependency relationships among tasks, human resources and level of effort

ID	Work Package	Time (h)	Prelations
1. Preliminary design			
1.	Preliminary design	30	
2. Structure and mechanics			
2.1.	Structure	6	BF - 1
2.2.	Deployments	6	BF - 2.1.
2.3.	Thermal protection	9	BF - 2.2.
2.4.	Commercial availability	12	BF - 2.3.
2.5.	Choose option	6	BF - 2.4.
3. Electrical Power System			
3.1.	Solar arrays	9	BF - 1, BB - 5, 6
3.2.	Batteries	9	BF - 1, BB - 5, 6
3.3.	Power management	12	BF - 3.1, 3.2
3.4.	Commercial availability	15	BF - 3.3.
3.5.	Choose option	9	BF - 3.4.
4. Propulsion Systems			
4.1	Motivations	12	BF - 1, BB - 2
4.2	Commercial availability	15	BF - 5
4.3	Choose option	9	BF - 4.2.
5. Payloads			
5.1.1.	Data Handling Systems	15	BF - 1
5.1.2.	Antenna	12	BF - 1, FF - 5.1.1.
5.2.	Commercial availability	15	BF - 5.1.
5.3.	Choose option	15	BF - 5.2.
6. AOCS			
6.1.	Attitude determination	9	BF - 1, BB - 2
6.2.	Attitude control	9	BF - 1, BB - 2
6.3.	Orbital Control	12	BF - 1, BB - 4
6.4.	Commercial availability	20	BF - 5
6.5.	Choose option	4	BF - 6.4.

Table 1.1: Prelations and Time

2 Gantt of the section

3 Satellite design

3.1 Structure and mechanics

The design and operation of a CubeSat is a complex process that must be completed keeping in mind the different subsystems that it has as well as the role they will play during the lifetime of the mission. And since these systems will operate in space, they have to be prepared and certified to withstand extreme temperature and radiation conditions.

The satellite used by Astrea must have high compatibility between all the systems to avoid potential problems and has to be tested (either all the systems together or one by one) and ensure their correct functioning. Given that the lifetime of the mission should be greater than four years, the critical systems such as the solar arrays, batteries and antennas should be fully operational until the end of the mission.

3.1.1 Structure

The mission of the structure is to sustain and protect all the electronic devices carried by the satellite in order to fulfill the mission requirements. In order to ensure that all the electronic and mechanic systems can be mounted upon the structure, a high compatibility between these systems is required. Given that the configuration of the CubeSat is not as common as other configurations of the current commercial or operational CubeSats, due to the mission of the project is to relay fast and reliable communication with the ground station and the other satellites, it is a really important point that the structure is highly flexible regarding the arrangement of the subsystems that it carries.

The structure chosen is manufactured by Innovative Solutions In Space (ISIS). Among its features it is worth mentioning that it can withstand the high range of temperature it will face in the space (from -40°C to 80°C) and it is highly compatible; almost every physical system used can be placed within the structure or on its faces (such as the antennas or the solar arrays). Finally, the mass of the structure is relatively low, and given that the mass of the other subsystems is not their best feature, it is plus point.

3.1.2 Thermal protection

The thermal protection system consists of various insulating materials that aim to protect the CubeSat from potential thermal shocks. The satellite must remain within an optimal range of temperature, despite of the variation of the external temperature, in order to work properly.

3.1 Structure and mechanics

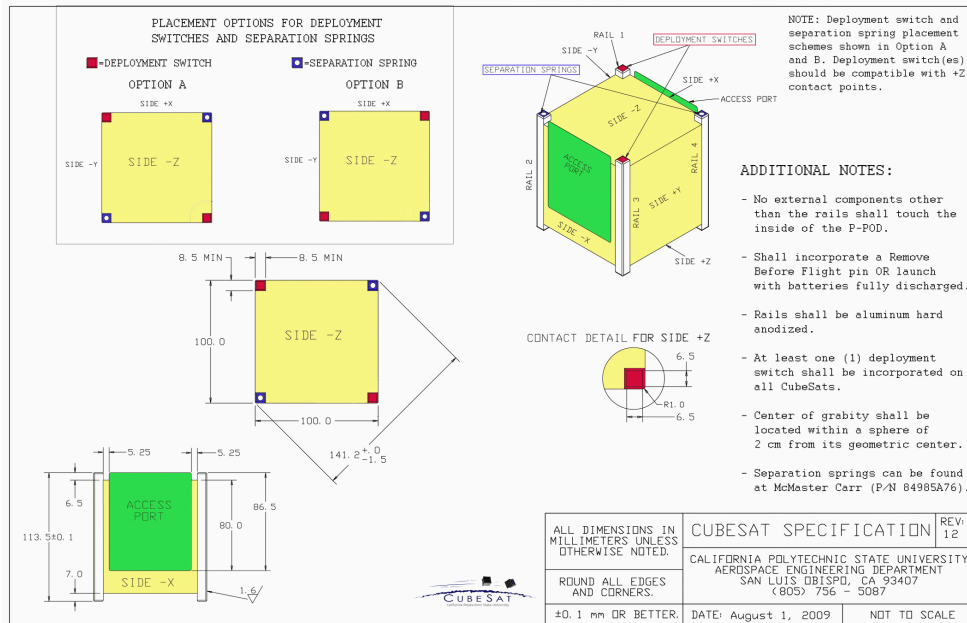


Figure 3.1: Dimensions of a 1U CubeSat

[?]

Operating in space, the CubeSat is vulnerable to suffer extreme temperatures, both below zero and above zero, and thermal protection must guarantee that all subsystems are protected. Furthermore, the thermal protection system should also dissipate the heat produced by the other systems.

Currently, the most used element as thermal protection in the aerospace industry is the multilayer insulation (MLI), a set of multiple thin insulation layers. The MLI fulfills all the requirements that were previously stated and its main objective is to reduce the heat generated by radiation since the heat generated by convection or conduction does not have such a high impact on the on-board systems.

After a market study, Dunmore Aerospace company has been chosen to provide us its MLI product. Specially, the product is the Dunmore Aerospace Satkit and it is made for small satellites for low earth orbit.

3.1.3 Study of the commercial available options and options chosen

A broad marked study is needed since all the options have to be considered. For this reason, and with the aim to show all the information and features of each system that has been considered in this section, the table 3.10 is presented below.

3.2 Electrical Power System

Brand and model	Features	Total price (€)
Structure		
ISIS 3U structure	Low mass (304.3g) Highly compatible High temperature range	3900
Gomspace GOMX-Platform	High mass (1500g) Comes fully equipped (basic systems) High temperature range	TO REQUEST!
Thermal protection		
Dunmore Aerospace Satkit	Lightweight Durability Made for small satellites	TO REQUEST!
Dupont Kapton Aircraft Thermal	Lightweight Durability Non-flammable	TO REQUEST!

Table 3.1: Options studied

Finally, the options chosen are presented in the table 3.11.

System	Brand and model	Price per unit (€)	N. of units
3U Structure	ISIS	3900	1
Thermal Protection	Dunmore Satkit	TO REQUEST	1

Table 3.2: Options chosen

3.2 Electrical Power System

The electric power system of the satellite must provide and manage the energy generated efficiently in order to have all the systems operating under normal conditions during the lifetime of the mission. The EPS of the Cubesat is, probably, the most fundamental requirement of the satellite, since its failure would result in a mission failure.

The energy collection system and the power management and collection systems compose the EPS and their role is to control and distribute power to the Cubesat, to supply a continuous source of electrical power during the length of the mission, to protect the satellite against electrical bus failiures and to monitor and communicate the status of the EPS to the on-board computer.

3.2 Electrical Power System

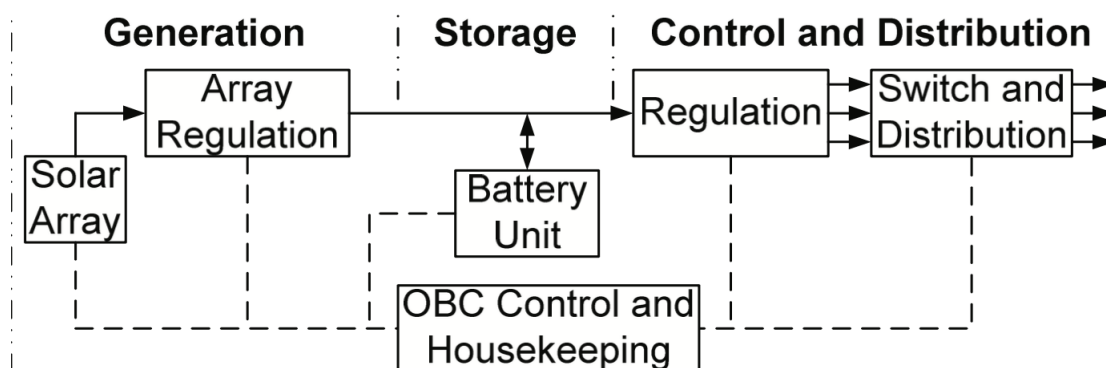


Figure 3.2: Basic schematics of the EPS

[?]

3.2.1 Estimation of the power required

To select the adequate electrical power systems it is essential that the power consumed by the CubeSat is known *a priori*. Thus, to select the solar arrays and batteries, as well as the power management system, an estimation of the power consumed has to be made.

The vast majority of the time the satellite will work under typical operation conditions. However, the estimation of the power consumption provided in the table 3.3 has been made for typical-high conditions in order to have a power margin and a more reliable estimation.

System (number of units)	Typical power consumption per unit (W)
Payload	
Patch antenna (4)	6
Turnstile antenna (2)	6
Payload power consumption	36
Electrical Power System	
NanoPower P60 Power Module (1)	2
Battery (2)	-
Solar arrays (4)	-
EPS power consumption	2
Data Handling Systems	
Transceiver (4)	6
DHS power consumption	X
Propulsion and ACDS	
Thruster (1)	20
Turnstile antenna (2)	6

3.2 Electrical Power System

OACDS power consumption	X
Estimated total power consumption	45

Table 3.3: Estimation of the power consumption under typical working conditions

3.2.2 Solar arrays

Given that the space of a 3U CubeSat is very limited, the primary source of electrical power has to be photovoltaic cells. The photovoltaic cells will collect and convert the energy of the sun into electrical energy and they have to be correctly selected to prevent failure given their importance.

The solar arrays used must have a decent efficiency and capacity to collect the energy from the sun, have to keep their mass relatively low, must have a protective radiation shield to ensure their full efficiency for at least 4 years, a proper deployment system, the ability to withstand space conditions and also must be highly compatible with all the other systems used, especially the power management system (the *NanoPower P60*).

The option selected for the mission is a set of deployable solar panels provided by EXA (Agencia Espacial Civil Ecuatoriana). These solar arrays fulfill all the requirements mentioned above: they are low mass (135g per unit), they have a protective radiation shield (NEMEA Anti Radiation Shield protects the solar panels of EM, High Gamma, X-Ray, Alfa, Beta and low neutron radiation) they can withstand a very high temperature range (from -80 to 130°C) ensuring that they can operate in space, they have a gentle release and deployment system with artificial muscles (developed by EXA) and they provide a power of 16.8W each (19.2V@0.5A).

Every cubesat will come with 4 deployable solar panels providing it with 67.2W of power, approximately, to supply peak demands during the lifetime of the mission. Additionally, it is worth mentioning that these solar arrays are compatible with the hardware used (the structure and the power management system).

3.2.3 Power management system

The role of the power management system is to distribute the power and supply the energy to the different systems used in the CubeSat. Since the systems of the CubeSat have different power and energy needs, the power management system has to be highly compatible and have a number of buses high enough to supply the different voltage and intensity required to the systems.

3.2 Electrical Power System

The selected option for the mission is the *NanoPower P60* by *Gomspace*, a high-power EPS for small satellites that comes with 1 motherboard, 1 ACU module (Array Conditioning Unit) and 1 PDU (Power Distribution Unit), allowing multiple configurations in just one motherboard; saving a lot of space.

The motherboard supports up to 4 ACU and PDU modules and has different regulated outputs (3.3V and 5V). It means that with one single motherboard, several conditioning and distributing units can be connected.

The ACU module 6 different inputs per unit with a high voltage solar input (up to 16V or 32V). Additionally, each input can withstand a maximum current of 2A and current and voltage inputs are measured on each input channel and the measurements can be communicated to the onboard computer.

The PDU module has 9 different outputs per unit that are highly configurable. Each module has 3 configurable output voltages (3.3V, 5V, 8V, 12V, 18V, 24V) and each of the outputs can withstand a maximum current of 1A or 2A (programmable). Additionally, like the ACU module, current and voltage outputs are measured on each output channel and can be effectively communicated to the onboard computer.

All these features make the *NanoPower P60* a very efficient and configurable power management unit that fulfills the mission requirements. Furthermore, given this capacity to configure each input and output channel and the high number of channels that it has, the compatibility between all the systems used in the satellite is ensured.

3.2.4 Batteries

Batteries are essential for a proper mission operation. They will provide the spacecraft subsystems with the power needed when the solar arrays are working less efficiently or not properly. Astrea is looking for decent capacity batteries that provide a slightly high typical energy and power supply, since all the systems will not usually operate under peak conditions. Additionally, through the lifetime of the mission, the solar arrays will face an important unfavorable condition; in the worst case scenario, the satellite will be in the dark during half of the time of the orbit. So, it is clear that the batteries are a critical system of the CubeSat

Among all the commercial options, Astrea has chosen the *BA01/D* batteries manufactured by *EXA-Agencia Espacial Civil Ecuatoriana*. The CubeSat will have two of these batteries, with a total capacity of 28800mAh or 106,4Wh. Each battery has a total of 16 cells, highly

3.2 Electrical Power System

stackable and with a very low mass (155g per unit). They also come with unique thermal transfer bus, that will transfer the heat of the other subsystems to the batteries to keep their temperature under efficient working conditions.

The output voltage can be configured (3.7V and 7.4V) and they are perfectly compatible with the solar arrays. Furthermore, they come with a protective radiation shield (NEMEA) that ensures at least 4 years working under full efficiency conditions in a LEO. It is also worth mentioning that if the company that will assemble the CubeSat faces problems during this part of the process, the batteries can be customized by contacting EXA.

3.2.5 Study of the commercial available options and options chosen

A broad marked study is needed since all the options have to be considered. For this reason, and with the aim to show all the information and features of each system that has been considered in this section, the table 3.4 is presented below.

Brand and model	Features	Total price (€)
Solar arrays		
EXA-Agencia Espacial Ecuatoriana	Total power of 67.2W (4units) Mass of 270g (p.unit) Included thermal protection At least 4 years lifetime	17000
ISIS	Total power of 30W (4units) Mass of 150g (p.unit) No thermal protection At least 2 years lifetime	TO REQUEST!
Power management		
Crystalspace P1 Vasik	Mass of 80g Full redundancy Low volume 6x outputs Up to 10W input High temperature range	TO REQUEST!
Gomspace NanoPower BP4	Mass of 176g 9x configurable outputs 6x inputs per module EMI shielding High temperature range	TO REQUEST!

3.3 Payload

Batteries		
Gomspace NanoPower BP4	Total capacity of 77Wh (2u) Automatic heat regulation Highly stackable Mass of 270g (p.unit)	TO REQUEST!
EXA-Agencia Espacial Ecuatoriana	Total capacity of 106.4Wh (2u) Automatic heat regulation Highly stackable Total mass of 155g	6300

Table 3.4: Options studied

Finally, the options chosen are presented in the table 3.5.

System	Brand and model	Price per unit (€)	N. of units
Solar arrays	EXA	17000	4
Batteries	EXA	6300	2
Power Management	Gomspace NanoPower P60	TO REQUEST!	

Table 3.5: Options studied

3.3 Payload

3.3.1 Antennas

The antennas are an essential part of the communication subsystem and their principal role is to transmit and receive the data. In order to provide fast and reliable communication, several options have been studied and the solution is presented on the lines below.

3.3.1.1 Patch antenna

FALTA DESCRIPCIÓN

Patch antenna	
Features	Value
Bands	L,S,C,X
Frequency range	1-12 GHz
Bandwidth	20 MHz
Gain	6 dBi
Polarization	Circular

3.3 Payload

Maximum power consumption	10 W
Impedance	50 Ohms
Operational temperature range	-65°C to +100°C
Mass	<250 grams

3.3.1.2 Turnstile antenna

FALTA DESCRIPCIÓN

Nando, ¿esto se puede arreglar un poco?

The frequency range is one of the most important parameters, because it must take into account satellite-satellite and satellite-earth communication. The initial requirement of the antenna frequency range is that it should be between 1-10 GHz. This is due to limitations in satellite-ground communication due to atmospheric conditions. Finding an antenna that meets this stringent requirement is very complicated, and a margin must be given to find an optimal market option.

The bandwidth is the frequency range where the highest power of the signal is found. The higher this bandwidth the better performance we will have.

The gain of an antenna is the ratio between the power density radiated in one direction and the power density that would radiate an isotropic antenna. The best option is to have a high gain.

Polarization is the orientation of the electromagnetic waves when leaving the antenna. There are three types of polarization: linear, circular and elliptical. For better performance, an antenna that receives and an antenna that transmits must have the same polarization. In project case, the best option is circular polarization because it is able to keep the signal constant regardless of the appearance of different problems such as movement with respect to the ground station.

The weight of the antennas should be as small as possible because the total weight of the cubesat should not exceed 4 kg. Most of the antennas of the market have a similar weight and does not cause us an extra problem when choosing the antenna of the project.

The power consumption parameter is an important requirement because most of the power is consumed by the different subsystems. The stage of greater power consumption due to

3.3 Payload

the antenna corresponds to its deployment, while once it is deployed, consumption is greatly reduced. In most cases, the power required for deployment ranges from 2-10 W.

The operational temperature range is important to the correct work of the antenna, because if the antenna was in a temperature outside this range, it would not be able to perform the communication of optimal form. An habitual temperature range use to be between XXXX

After a market study, the antennas chosen to perform the communication have been a Microstrip Patch Antenna developed by Antenna Development Corporation and a turnstile antenna ANT430. On the back and lower face of the cubesat will be implemented turnstile antennas, while on the lateral sides will be implemented the antennas patch.

On the lower face of the cubesat is necessary to use an antenna turnstile because a thruster must be incorporated.

The following table shows the main parameters of those antennas.

Turnstile antenna	
Features	Value
Frequency range	400-480 MHz
Bandwidth	5 MHz
Gain	1.5 dBi
Polarization	Circular
Maximum power consumption	10 W
Impedance	50 Ohms
Operational temperature range	-40°C to +85°C
Mass	30 grams

Table 3.7: Main features of the turnstile antenna

3.3.2 Data Handling Systems

The communication system allows us to realize the reception and trasmission of data, voice signals, etc. It consists of a group of transponders, that are the combination of a transmitter and a receiver and whose functions are receiving, separating, amplify, process, reamplify and retransmit signals.

The telemetry subsystem analyses the information about the ground station and other sensors of the satellite in order to monitor conditions on board. It allows report to ground station about the conditions of the on board systems.

3.4 Propulsion Systems

The command and control subsystem allows the ground station to control the satellite.

3.3.3 Study of the commercial available options and options chosen

System	Brand and model	Price per unit (€)	N. of units
Patch antenna	ADC, Microstrip	TO REQUEST!	4
Turnstile antenna 1	Gomspace, ANT430	TO REQUEST	2
TRANSCEIVER	EMPTY	TO REQUEST	1
DHS	EMPTY	20000	1

Table 3.8: Options studied

3.4 Propulsion Systems

3.4.1 Requirements

There is a big risk of a collision with space debris while a spacecraft is operating in Low Earth Orbits. The Inter-Agency Space Debris Coordination Committee recommended to the United Nations (section 5.3.2 ‘Objects Passing Through the LEO Region’): “Whenever possible space systems that are terminating their operational phases in orbits that pass through the LEO region, or have the potential to interfere with the LEO region, should be de-orbited (direct re-entry is preferred) or where appropriate manoeuvred into an orbit with a reduced lifetime. Retrieval is also a disposal option.” and “A space system should be left in an orbit in which, using an accepted nominal projection for solar activity, atmospheric drag will limit the orbital lifetime after completion of operations. A study on the effect of post- mission orbital lifetime limitation on collision rate and debris population growth has been performed by the IADC. This IADC and some other studies and a number of existing national guidelines have found 25 years to be a reasonable and appropriate lifetime limit.”

Thus, a proper propulsion system is needed both for maintaining the satellite’s orbit and for de-orbiting after the mission’s lifetime.

Given the size of the CubeSat, not many effective options are available and a committed solution has to be found in order to follow the recommendations by the IADC.

3.4.2 Orbit decay

Orbit decay prediction powered by the Bureau of Meteorology by the Australian Government.

To calculate the orbit decay the following parameters are used:

3.4 Propulsion Systems

Solar Radio Flux at 10.7cm (F10.7). It is a clear indicator of solar activity and has proven very valuable in forecasting space weather. The extreme UV that impact the ionosphere also modify the upper atmosphere, thus F10.7 data is needed to account for these variations. The value used in this calculation is: 79.54. REF: <http://www.spaceweather.gc.ca/solarflux/sx-5-mavg-en.php>

Cubesat mass of up to 4kg.

The K-index, and by extension the Planetary K-index, are used to characterize the magnitude of geomagnetic storms. Kp is an excellent indicator of disturbances in the Earth's magnetic field and is used by SWPC to decide whether geomagnetic alerts and warnings need to be issued for users who are affected by these disturbances.

The principal users affected by geomagnetic storms are the electrical power grid, spacecraft operations, users of radio signals that reflect off of or pass through the ionosphere, and observers of the aurora.

The geomagnetic index used in this calculation is: 12.

RE-ENTRY EVERY 106.25 DAYS!

Calculations based on day 100, with an altitude of 400Km. Already lost 100Km:

3.4.3 Thrusters

Thruster is a main part of the structure because it is needed to allow the satellite to realise different maneuvers how incorporate it adequately to the orbit after the deployment of the rocket, can obtain the optimal orientation or to maintain the satellite in the orbital and avoid its fallen.

The main parameters that must consider are thrust, total specific impulse, power required, weight of the propulsion subsystem and its volume.

At the moment, the most used and more modern thrusters for satellites are: ionic, pulsed plasma, electrothermal and green monopropellant thrusters. An important aspect to consider is that we are interested in is reducing the mass required although this will cause minor accelerations than conventional engines but it will be suitable for small satellites.

After a market study, the best two options to consider are the green monopropellant thruster BGT-X5 and the ion thruster BIT-1, both from Busek company. These two thrusters are among the most used in the aerospace industry for small satellites. The main difference between both are the thrust and the specific impulse. On the one hand, the BIT-1 thruster provides a lower

3.4 Propulsion Systems

thrust but with a high specific impulse. On the other hand, BGT-X5 thruster provides a high thrust, around 0.5 N but with a low specific impulse.

Finally, BGT-X5 has been chosen as the CubeSat thruster. With the high thrust and delta V that BGT-X5 provides, the CubeSat will be able to carry out the necessary actions to keep the satellite in orbit, to relocate the satellite or to change its orbit.

The following table shows the main parameters of this thruster.

BGT-X5	
PARAMETERS	VALUE
Total thruster power	20 W
Thrust	0.5 N
Specific impulse	225 s
Thruster Mass	1500 g
Input voltage	12 V
Delta V	146 m/s

3.4.4 Study of the commercial available options

A broad market study is needed since all the options have to be considered. For this reason, and with the aim to show all the information and features of each system that has been considered in this section, the table 3.10 is presented below.

Brand and model	Features	Total price (€)
Propulsion		
Busek ion thruster BIT-1	Volume 1/2 U High Isp (2150 s) Low thrust (100 μ N)	TO REQUEST
Busek BGT-X5	Volume 1 U High thrust (0.5 N) High delta V (146 m/s)	TO REQUEST!

Table 3.10: Options studied

Finally, the option chosen is presented in the table 3.11.

System	Brand and model	Price per unit (€)
BGT-X5	Busek Company	TO REQUEST)

3.5 Communication module

Table 3.11: Options chosen

3.5 Communication module

100kbps:

1mbps:

¿solo 9600bps?:

Links interesantes universidades:

3.6 Attitude and Orbital Control Systems

Attitude and orbital control subsystem is needed to enable the satellite to keep a specific position within its orbit and to control the antennas in order to remain oriented to assigned area, because the satellite tends to change its orientation due to torque. The AOCS receives telecommands from the central computer and acquires measurements (satellite attitude and orbital position) from sensors. We will also refer to the attitude control as ADACS (Attitude Determination and Attitude control system).

3.6.1 Attitude Determination

EMPTY

3.6.2 Attitude Control

EMPTY

3.6.3 Orbital Control

Thrusters

3.6.4 Study of the commercial available options

Brand and model	Features and description	Money (€)
Solar Panels		
Fabricant 1	EMPTY	2000000
Chuscas 1	EMPTY	20000
Truñaas 1	EMPTY	20000

3.7 Link Budget

Cuescas 1	EMPTY	20000
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Table 3.12: Options studied

3.7 Link Budget

Astrea constellation main satellite must be able to establish three different telecommunications link:

- Space to Ground link.
- Space to Space link between Astrea satellites.
- Space to Space link between client and Astrea satellites.

The link budget calculations are mostly calculated using the following formula: [?].

3.7.1 Communications Basics

[?] When evaluating a wireless link, the three most important questions to be answered are:

1. How much radio frequency (RF) power is available?
2. How much bandwidth is available?
3. What is the required reliability (as defined by Bit Error Rate, or BER)?

The upper limit in terms of data rate is given by Shannon's Channel Capacity Theorem:

$$C = B \log_2(1 + S/N) \quad (3.1)$$

where:

C = channel capacity (bits/s)

B = channel bandwidth (Hz)

S = signal strength (watts)

N = noise power (watts)

Transmission Losses In any satellite transmission, there are always losses from various sources. Some of those losses may be constant, others are dependent of statistical data and others vary with the weather conditions, especially with rain.

3.7 Link Budget

TRANSMISSION LOSSES	PROPAGATION LOSSES	FREE SPACE LOSSES		
		ATMOSPHERIC LOSSES	Ionospheric effects	Faraday rotation Scintillation effects
			Tropospheric effects	Attenuation
				Rain attenuation
				Gas absorption
				Depolarization
			Sky noise	
		Local effects		
	POINTING LOSSES			
	LOCAL LOSSES	EQUIPMENT LOSSES	Feeder losses	
		?????		
		ENVIRONMENT LOSSES		

Figure 3.3: Principal losses in the received signal [?]

3.7.2 Propagation losses

3.7.2.1 Free Space Losses

Range and Path Loss Another key consideration is the issue of range. As radio waves propagate in free space, power falls off as the square of range. For a doubling of range, power reaching a receiver antenna is reduced by a factor of four. This effect is due to the spreading of the radio waves as they propagate, and can be calculated by:

$$L = 20 \log_{10}(4\pi D/\lambda) \quad (3.2)$$

where:

D = the distance between receiver and transmitter

λ = free space wavelength = c/f

c = speed of light ($3 \times 10^8 m/s$)

f = frequency (Hz)

3.7 Link Budget

3.7.2.2 Atmospheric Losses

This kind of losses derives from the absorption of energy by atmospheric gases. They can assume two different types:

- Atmospheric attenuation.
- Atmospheric absorption.

The major distinguishing factor between them is their origin. Attenuation is weatherrelated, while absorption comes in clear-sky conditions. Likewise, these losses can be due to ionospheric, tropospheric and other local effects. [?]

Ionospheric Effects All radio waves transmitted by satellites to the Earth or vice versa must pass through the ionosphere, the highest layer of the atmosphere, which contains ionized particles, especially due to the action of sun's radiation. Free electrons are distributed in layers and clouds of electrons may be formed, originating what is known as travelling ionospheric disturbances, what provoke signal fluctuations that are only treated as statistical data. The effects are:

- **Polarization rotation:** When a radio wave passes through the ionosphere, it contacts the layers of ionized electrons that move according to the Earth's magnetic field. The direction these electrons move will no longer be parallel to the electric field of the wave and therefore the polarization is shifted, in what is called Faraday rotation (θ_F). ;
- **Scintillation effects:** Differences in the atmospheric refractive index may cause scattering and multipath effect, due to the different directions rays may take through the atmosphere. They are detected as variations in amplitude, phase, polarization and angle of arrival of the radio waves. It is often recommended the introduction of a fade margin so atmospheric scintillation can be a tolerated phenomenon.;
- Absorption;
- Variation in the direction of arrival;
- Propagation delay;
- Dispersion;
- Frequency change

These effects decrease usually with the increase of the square of the frequency and most serious ones in satellite communications are the polarization rotation and the scintillation effects, and those are the ones that will be treated in this dissertation. [?]

3.7 Link Budget

Tropospheric Effects [?] Troposphere is composed by a miscellany of molecules of different compounds, such as hail, raindrops or other atmospheric gases. Radio waves that pass by troposphere will suffer their effects and will be scattered, depolarized, absorbed and therefore attenuated.

Attenuation: As radio waves cross troposphere, radio frequency energy will be converted into thermal energy and that attenuates signal.

Rain attenuation: Ground stations had been chosen in order that the attenuation caused by rainfall will be very punctual. Also, the fact that there are three ground stations makes really difficult that a satellite can not communicate to the ground in all the orbit period.

Gas absorption: Under normal conditions, only oxygen and water vapour have a significant contribution in absorption. Other atmospheric gases only become a problem in very dry air conditions above 70 GHz. Thereby, losses caused by atmospheric absorption vary with frequency and the collection of data already received allows the elaboration of the graphic that follows:

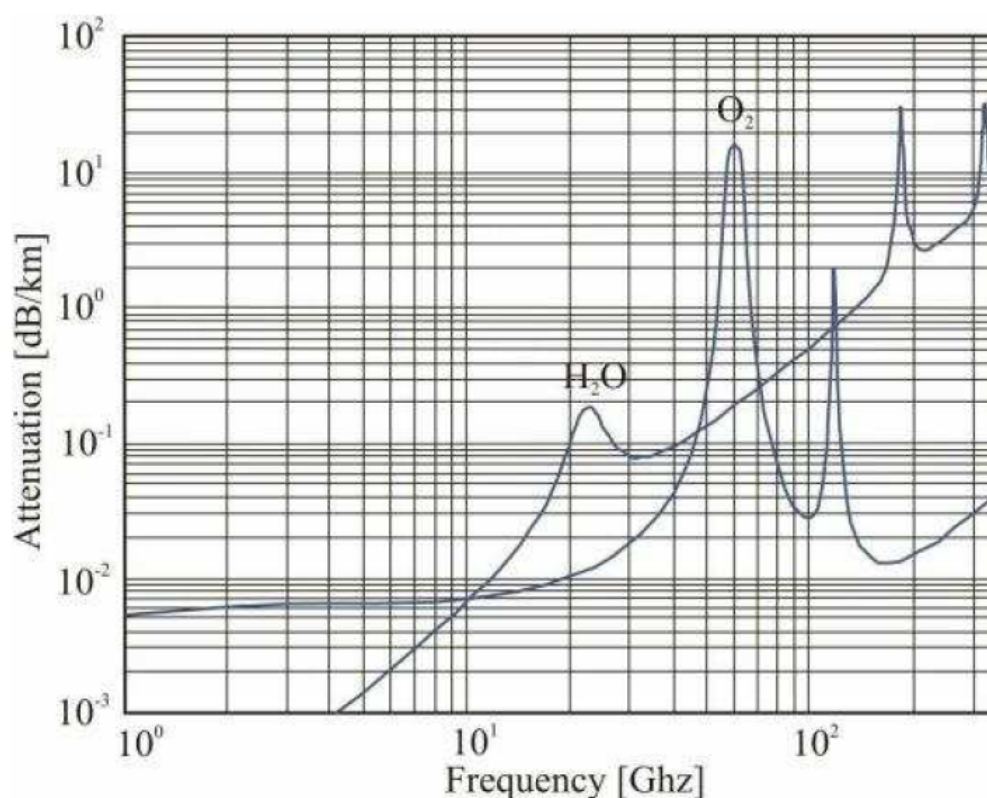


Figure 3.4: Specific attenuation for different frequencies [?]

Local Effects

3.7 Link Budget

3.7.2.3 Pointing Losses

3.7.2.4 Multipath and Fade Margin

Multipath occurs when waves emitted by the transmitter travel along a different path and interfere destructively with waves travelling on a direct line-of-sight path. This is sometimes referred to as signal fading. This phenomenon occurs because waves travelling along different paths may be completely out of phase when they reach the antenna, thereby cancelling each other.

The amount of extra RF power radiated to overcome this phenomenon is referred to as fade margin. The exact amount of fade margin required depends on the desired reliability of the link, but a good rule-of-thumb is 20dB to 30dB.

3.7.3 Local Losses

3.7.3.1 Equipament Losses

3.7.3.2 Environment Losses

3.7.4 Modulation Technique

Modulation technique is a key consideration. This is the method by which the analogue or digital information is converted to signals at RF frequencies suitable for transmission. Selection of modulation method determines system bandwidth, power efficiency, sensitivity, and complexity. Most of us are familiar with Amplitude Modulation (AM) and Frequency Modulation (FM) because of their widespread use in commercial radio. Phase Modulation is another important technique. It is used in applications such as Global Position System (GPS) receivers and some cellular telephone networks. [?]

For the purposes of link budget analysis, the most important aspect of a given modulation technique is the Signal-to- Noise Ratio (SNR) necessary for a receiver to achieve a specified level of reliability in terms of BER.

A graph of E_b/N_o vs BER is shown in Figure 3.5. E_b/N_o is a measure of the required energy per bit relative to the noise power. Note that E_b/N_o is independent of the system data rate. In order to convert from E_b/N_o to SNR , the data rate and system bandwidth must be taken into account as shown below:

$$SNR = (E_b/N_o)(R/B_T) \quad (3.3)$$

3.7 Link Budget

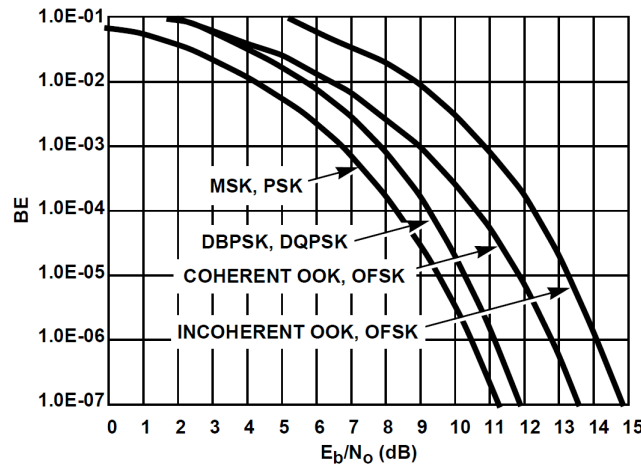


Figure 3.5: Probability of bit error for common modulation methods[?]

where:

E_b = Energy required per bit of information

N_o = thermal noise in 1Hz of bandwidth

R = system data rate

B_T = system bandwidth

3.7.5 System Noise

Channel Noise All objects which have heat emit RF energy in the form of random (Gaussian) noise. The amount of radiation emitted can be calculated by [?]:

$$N = kTB \quad (3.4)$$

where:

N = noise power (watts)

k = Boltzman's constant($1.38 \times 10^{-23} J/K$)

T = system temperature, usually assumed to be 290K

B = channel bandwidth (Hz)

This is the lowest possible noise level for a system with a given physical temperature. For most applications, temperature is typically assumed to be room temperature (290K). Equations 3.1 and 3.4 demonstrate that RF power and bandwidth can be traded off to achieve a given performance level (as defined by BER).

3.7 Link Budget

3.7.6 Link Budget Calculation

Methodology From the expected requirements fixed on the Project Charter, general radio systems parameters will be computed, in order to have a reference to look for the best communications system on board the Astrea satellites.

$$EIRP = P_T - L_T - G_T$$