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Cubesat Constellation Astrea

Report

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1 Satellite design planning




















Interdependency relationships among tasks, human resources and level of effort

ID	Work Package	Time (h)	Prelations
1. Preliminary design			
1.	Preliminary design	30	
2. Structure and mechanics			
2.1.	Structure	6	BF - 1
2.2.	Deployments	6	BF - 2.1.
2.3.	Thermal protection	9	BF - 2.2.
2.4.	Commercial availability	12	BF - 2.3.
2.5.	Choose option	6	BF - 2.4.
3. Electrical Power System			
3.1.	Solar arrays	9	BF - 1, BB - 5, 6
3.2.	Batteries	9	BF - 1, BB - 5, 6
3.3.	Power management	12	BF - 3.1, 3.2
3.4.	Commercial availability	15	BF - 3.3.
3.5.	Choose option	9	BF - 3.4.
4. Propulsion Systems			
4.1	Motivations	12	BF - 1, BB - 2
4.2	Commercial availability	15	BF - 5
4.3	Choose option	9	BF - 4.2.
5. Payloads			
5.1.1.	Data Handling Systems	15	BF - 1
5.1.2.	Antenna	12	BF - 1, FF - 5.1.1.
5.2.	Commercial availability	15	BF - 5.1.
5.3.	Choose option	15	BF - 5.2.
6. AOCS			
6.1.	Attitude determination	9	BF - 1, BB - 2
6.2.	Attitude control	9	BF - 1, BB - 2
6.3.	Orbital Control	12	BF - 1, BB - 4
6.4.	Commercial availability	20	BF - 5
6.5.	Choose option	4	BF - 6.4.

Table 1.1: Prelations and Time

2 Gantt of the section

[illegible]

Proyecto: planning Fecha: mié 19/10/16	Tarea		Resumen del proyecto		Tarea manual		solo el comienzo		Fecha límite	
	División		Tarea inactiva		solo duración		solo fin		Progreso	
	Hito		Hito inactivo		Informe de resumen manual		Tareas externas		Progreso manual	
	Resumen		Resumen inactivo		Resumen manual		Hito externo			

3 Satellite design

3.1 Structure and mechanics

The design and operation of a CubeSat is a complex process that must be completed keeping in mind the different subsystems it has and their role within the satellite. Additionally, since these systems will operate in space, they have to be prepared to withstand extreme conditions and be certified to work properly under them. The satellite used by Astrea must have high compatibility between all the systems to avoid potential problems during operation and has to be tested (either all the systems together or one by one) and ensure their correct functioning.

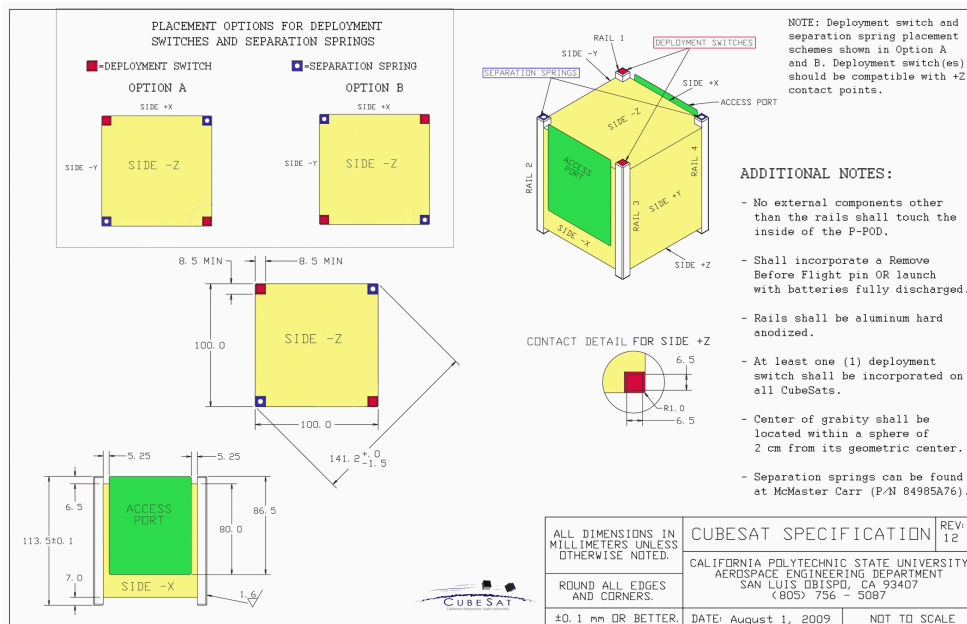


Figure 3.1: Dimensions of a 1U CubeSat [?]

3.1.1 Structure

The mission of the structure is to sustain and protect all the electronic devices carried by the satellite in order to fulfill the mission requirements. In order to ensure that all the electronic and mechanic systems can be mounted upon the structure, a high compatibility between these systems is required.

The structure used will be provided by Innovative Solutions In Space (ISIS). Among its features it is worth mentioning that it can withstand the high range of temperature it will face in the space (-40°C to 80°C) and it is highly compatible with almost every physical system that will be used. Additionally, it is a low mass structure (304.3g) and it can support different configurations within it.

3.2 Electrical Power System

Since the configuration within the CubeSat will not be as common as the other configuration within the current operational CubeSats, because the mission of our project is to relay fast and reliable communication with the ground station and the other satellites, it is a really important point that the structure is highly flexible regarding the arrangement of the subsystems that it will carry.

3.1.2 Deployments

EMPTY

3.1.3 Thermal protection

The thermal protection system protect the CubeSat from thermal shocks. The satellite must remain in a optimal range of temperatures, despite the external temperature. It consists of various insulating materials and thermal conductors in order to maintain it within acceptable temperatures.

3.1.4 Study of the commercial available options

EMPTY

System	Brand and model	Money (€)
3U Structure	ISIS	3900)
EMPTY 1	EMPTY	EMPTY
EMPTY 1	EMPTY	EMPTY

Table 3.1: Options studied

Of all the options in 3.5, we have chosen the following options.

3.2 Electrical Power System

The Electric Power System of the satellite must provide and manage the energy generated efficiently in order to have all the systems operating under normal conditions. The Electrical Power System of a Cubesat is, probably, the most fundamental requirement of the satellite payload, since its failure would result in the mission failure. The functions of the EPS are to control and distribute power to the Cubesat, to supply a continuous source of electrical power for through the lifetime of the mission, to protect the satellite against bus failiures and to monitor and communicate the system status to the on-board computer. The role of the EPS is very diverse and the following subsystems have to be analyzed in detail.

3.2 Electrical Power System

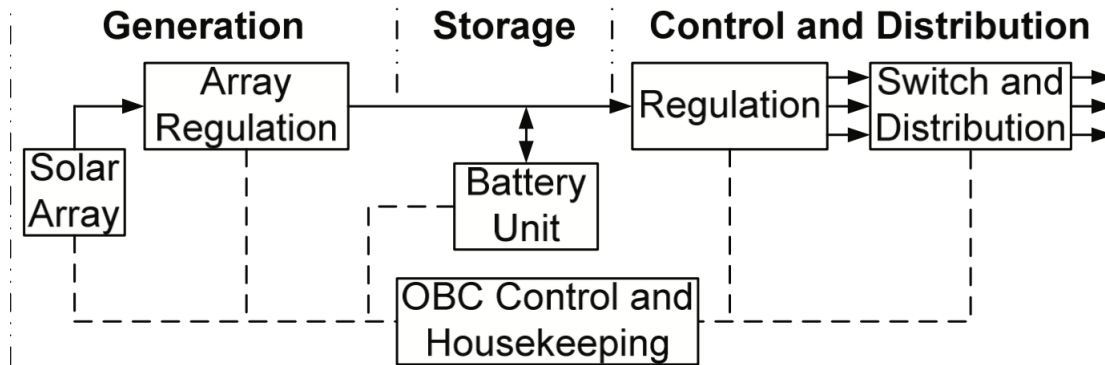


Figure 3.2: Basic schematics of the EPS [?]

3.2.1 Solar arrays

The primary source of electrical power has to be photovoltaic cells, given the size of the CubeSat. The photovoltaic cells will collect and convert the energy of the sun into electrical energy. Since they are main power source, they have to be correctly selected to prevent failure. Among the characteristics we seek for our mission, we are looking specially whether the solar cell has a decent amount of power collection, low mass, protective radiation shield, deployment system, great temperature range and compatibility with the other systems.

The selected option for the Astrea mission will be a set of deployable solar panels provided by EXA (Agencia Espacial Civil Ecuatoriana). These panels are low mass (135g each), have a protective radiation shield (NEMEA Anti Radiation Shield protects the solar panels of EM, High Gamma, X-Ray, Alfa, Beta and low neutron radiation), a very high temperature range (from -80 to 130°C), a gentle release and deployment with artificial muscles (developed by EXA) and provide a power of 16.8W each (19.2V@0.5A).

Every cubesat will have 4 deployable solar panels providing it with 67.2W of power to supply peak demands while it is operating. Additionally, it is worth mentioning that these solar arrays are compatible with the hardware used in each of the satellites.

3.2.2 Power management

The role of the power management systems is to distribute the power and supply the energy to the different systems used in the CubeSat.

3.2.3 Batteries

Batteries are essential for a proper mission operation. They will provide the spacecraft subsystems with the power needed when the solar arrays are working less efficiently or not properly. Astrea is looking for decent capacity batteries that provide a slightly high typical

3.3 Propulsion Systems

energy and power, since the subsystems will not usually operate under peak conditions. Through the lifetime of the mission, the solar arrays will face an important unfavorable condition; in the worst case scenario, the satellite will be in the dark during half of the time of the orbit. Thus, the batteries must store and supply the energy needed for this time.

Among all the commercial options, Astrea has chosen the *NanoPower BP4* batteries. The CubeSat will have two of these batteries, with a total capacity of 20800mAh or 77Wh. Each battery has a total of 4 cells, highly stackable and come with a temperature sensor and a heater that will automatically turn on if the temperature is worsening the whole operation of the system.

3.2.4 Study of the commercial available options

Several commercial options have to be studied. The table provided below organizes some information about the different options purchased.

System	Brand and model	Price per unit (€)
Solar arrays	EXA-Agencia Espacial Civil Ecuatoriana	17000
Batteries 1	NanoPower BP4 by Gomspace	TO REQUEST!
Power Management 1	NanoPower	TO REQUEST!

Table 3.2: Options studied

Of all the options in 3.5, we have chosen the following options.

3.3 Propulsion Systems

3.3.1 Requirements

There is a big risk of a collision with space debris while a spacecraft is operating in Low Earth Orbits. The Inter-Agency Space Debris Coordination Committee recommended to the United Nations (section 5.3.2 ‘Objects Passing Through the LEO Region’): “Whenever possible space systems that are terminating their operational phases in orbits that pass through the LEO region, or have the potential to interfere with the LEO region, should be de-orbited (direct re-entry is preferred) or where appropriate manoeuvred into an orbit with a reduced lifetime. Retrieval is also a disposal option.” and “A space system should be left in an orbit in which, using an accepted nominal projection for solar activity, atmospheric drag will limit the orbital lifetime after completion of operations. A study on the effect of post- mission orbital lifetime limitation on collision rate and debris population growth has been performed by the IADC.

3.3 Propulsion Systems

This IADC and some other studies and a number of existing national guidelines have found 25 years to be a reasonable and appropriate lifetime limit.”

Thus, a proper propulsion system is needed both for maintaining the satellite’s orbit and for de-orbiting after the mission’s lifetime.

Given the size of the CubeSat, not many effective options are available and a committed solution has to be found in order to follow the recommendations by the IADC.

3.3.2 Orbit decay

Orbit decay prediction powered by the Bureau of Meteorology by the Australian Government.

To calculate the orbit decay the following parameters are used:

Solar Radio Flux at 10.7cm (F10.7). It is a clear indicator of solar activity and has proven very valuable in forecasting space weather. The extreme UV that impact the ionosphere also modify the upper atmosphere, thus F10.7 data is needed to account for these variations. The value used in this calculation is: 79.54. REF: <http://www.spaceweather.gc.ca/solarflux/sx-5-mavg-en.php>

Cubesat mass of up to 4kg.

The K-index, and by extension the Planetary K-index, are used to characterize the magnitude of geomagnetic storms. Kp is an excellent indicator of disturbances in the Earth’s magnetic field and is used by SWPC to decide whether geomagnetic alerts and warnings need to be issued for users who are affected by these disturbances.

The principal users affected by geomagnetic storms are the electrical power grid, spacecraft operations, users of radio signals that reflect off of or pass through the ionosphere, and observers of the aurora.

The geomagnetic index used in this calculation is: 12.

RE-ENTRY EVERY 106.25 DAYS!

Calculations based on day 100, with an altitude of 400Km. Already lost 100Km:

3.3.3 Thrusters

Thruster is a main part of the structure because it is needed to allow the satellite to realise different maneuvers how incorporate it adequately to the orbit after the deployment of the rocket, can obtain the optimal orientation or to maintain the satellite in the orbital and avoid its fallen.

3.4 Communication module

The main parameters that must consider are thrust, total specific impulse, power required, weight of the propulsion subsystem and its volume.

At the moment, the most used and more modern thrusters for satellites are: ionic, pulsed plasma, electrothermal and green monoprop thrusters. An important aspect to consider is that we are interested in is reducing the mass required although this will cause minor accelerations than conventional engines but it will be suitable for small satellites.

After a market study, an ionic thruster has been elected how the best option. The causes of this election are that the volume of the all propulsion subsystem and its weight are very small, specific impulse is very high, thrust is acceptable and power required can be supplied by the solar panels.

The following table shows the main parameters of this thruster.

BIT-1 ION THRUSTER	
PARAMETERS	VALUE
Total thruster power	10 W
Thrust	100 μ N
Specific impulse	2150 s
Thruster Mass	53 g
Propellant mass flow	4.9 μ g/s
Grid input voltage	2 kV
Ion beam current	1.5 mA
Propellant utilization	41 percent
Energy Efficiency	27 percent

3.4 Communication module

100kbps:

1mbps:

¿solo 9600bps?:

Links interesantes universidades:

3.5 Attitude and Orbital Control Systems

Attitude and orbital control subsystem is needed to enable the satellite to keep a specific position within its orbit and to control the antennas in order to remain oriented to assigned

3.6 Payload

area, because the satellite tends to change its orientation due to torque. The AOCS receives telecommands from the central computer and acquires measurements (satellite attitude and orbital position) from sensors. We will also refer to the attitude control as ADACS (Attitude Determination and Attitude control system).

3.5.1 Attitude Determination

EMPTY

3.5.2 Attitude Control

EMPTY

3.5.3 Orbital Control

Thrusters

3.5.4 Study of the commercial available options

Brand and model	Features and description	Money (€)
Solar Panels		
Fabricant 1	EMPTY	2000000
Chuscas 1	EMPTY	20000
Truñaas 1	EMPTY	20000
Cuescas 1	EMPTY	20000

Table 3.4: Options studied

3.6 Payload

EMPTY

3.6.1 Communications systems

The communication system allows us to realize the reception and trasmission of data, voice signals, etc. It consists of a group of transponders, that are the combination of a transmitter and a receiver and whose functions are receiving, separating, amplify, process, reamplify and retransmit signals.

3.6 Payload

The telemetry subsystem analyses the information about the ground station and other sensors of the satellite in order to monitor conditions on board. It allows report to ground station about the conditions of the on board systems.

The command and control subsystem allows the ground station to control the satellite.

3.6.1.1 Data Handling Systems

EMPTY

3.6.1.2 Antenna

EMPTY

3.6.2 Study of the commercial available options

EMPTY

Brand and model	Features and description	Money (€)
Solar Panels		
Fabricant 1	EMPTY	2000000
Chuscas 1	EMPTY	20000
Truñaas 1	EMPTY	20000
Cuescas 1	EMPTY	20000

Table 3.5: Options studied

4 References