

# Space Debris Classification

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## 1 Introduction

This project is a study to attempt to create a model to classify a portion of debris orbiting Earth. The first section is an introduction describing the background material, what is space debris and why is it an issue. The second section describes the dataset that is used for the statistical model. The third section will discuss the statistical model itself and why it was chosen. The fourth section will look at the results obtained from the model and what conclusions can be drawn from the dataset and the model and the possible future work and implementations of this model.

## 2 Background

Ever since humans have been sending things into space we have not been cleaning up after ourselves. Sending things into space requires a lot of energy and cost, so bringing things back costs even more. The debris that is in space is the remains of dead satellites, rocket chassis, engines, components of space stations, and other miscellaneous items. Fortunately, some items that are close enough to Earth reenter the atmosphere and are burned up on reentry. But there are still many objects that will remain in orbit.

### 3 Methods and Data

Before going into the statistical analysis methods used, it is important to understand the dataset used for satellite tracking. The dataset used is a collection of objects orbiting in the Low-Earth orbit (LEO) region. There are many other regions including, geo-synchronous orbit (GEO), high earth orbit (HEO). Satellite tracking data is encoded in a two line element set. The two line element set encodes position and velocity of a satellite based on its orbital elements. Orbital elements is a coordinate system used to describe the shape of an orbit. The two element set also includes data such as identification number, security classification either classified or unclassified epoch launch dates, and atmospheric drag coefficients. The position and velocity of the objects are computed using simplified perturbation models. The simplified perturbation models are used to compute the orbital state of an object which taking into account the shape of the Earth, radiation, atmospheric drag, and gravitational influence from the Moon and Sun.

The dataset used for this project is provided by [tarck-sat.org](http://tarck-sat.org). This organization collects and maintains datasets of satellite position status and position data. The data is obtained from NORAD(North American Aerospace Defense Command) and NASA. The dataset will be analyzed using various methods. One interesting thing to look at is to see which countries have which objects. This is a good insight into what countries have what type of debris. The object type seen in this dataset are: debris, payload, rocket bodies, and others. Debris in this context consists of satellite and spacecraft fragments, payload is made up of capsules of supplies or satellites. Rocket bodies consist of discarded rockets that were used to launch objects into orbit.

Along with the object type the radar cross-section is an important metric to look at. The radar cross section is used to estimate the cross-sectional area of an object. For the purposes of this study, a larger radar cross section means that

an object is larger. Specific values for the radar cross section is not provided due to security reasons or lack of accuracy from the ground station sensors. Radar cross section is divided into three categories, small, medium, and large. Small are objects  $< .1m^2$ , medium are objects between  $.1m^2$  and  $1m^2$ , and large objects are  $> 1m^2$ .

The given dataset also provides orbital element data. The orbital element data provided are: apogee, perigee, inclination, and Period. The apogee and perigee are the maximum and minimum radii of an elliptical orbit, respectively. The inclination is the tilt angle of the orbit relative to the Earth's equator and the period is the time it takes for the object to orbit the Earth in one revolution.

The dataset provided will be used to create a classifier. This classifier will determine how much potential risk an object is. This risk metric is a function of two things, the mean orbit speed of an object and the radar cross section. The classifier works by determining that object with large cross sectional area and high velocities have the most risk of damaging or colliding with other objects than slower and smaller objects.

As mentioned earlier, the risk assessment will be based on the mean orbit speed, the mean orbit speed can be determined from the equation below.

$$v = \frac{2a}{T} \quad (1)$$

where T is the orbit period given by the dataset in seconds and a is the semimajor axis in kilometers, given in the equation below

$$a = \frac{r_p + r_a}{2} \quad (2)$$

Where  $r_p$  is perigee and  $r_a$  is apogee.

## 4 Results

## 5 Conclusion

This paper goes into the background information on space debris. Then the dataset and approaches were discussed. Then the results of the statistical analysis were discussed.

The future work of this project could include implementing more complicated classification models like using neural networks. Given the results of this project with the large amount of debris orbiting Earth, many people working on trying to mitigate the orbital debris problem.