

# Final Presentation

# SKULLD PROPELLION

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# Design Team Organization

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## Team



Cooper A. Drain

Systems Engineer



Benjamin B. Sheldon

Programmer



Biswas Poudel

Test Engineer



Shane P. Cullen

Structural Engineer



Kimiya Ghobadi

Systems Engineer



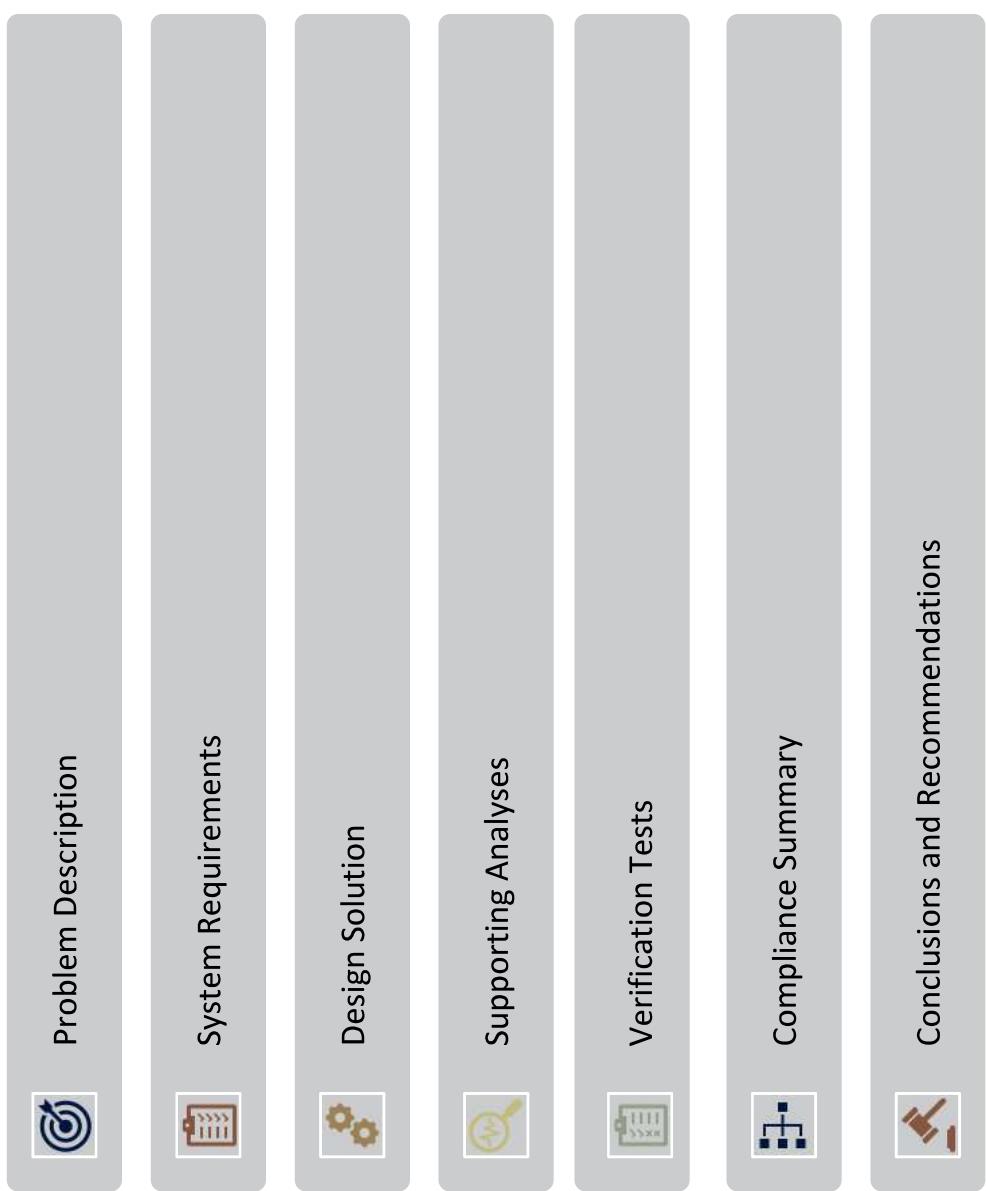
Andrew M. Allison

Instrumentation Engineer



McConnell E. Gou

# Agenda

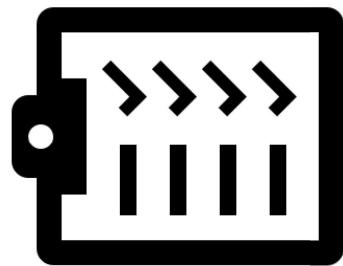


# Purpose & Problem Statement

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The current process for characterizing propellant at Embry-Riddle Aeronautical University is time-consuming, inefficient, and unreliable. To effectively characterize solid rocket propellant and cut costs, a strand burner with a new data acquisition system was designed.





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# System Requirements

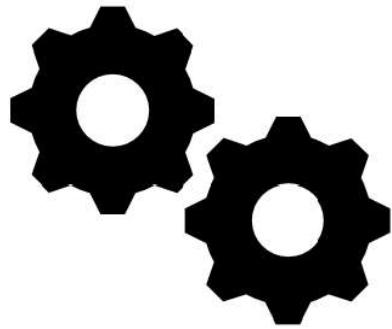


# System Requirements

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1. Cut cost of characterization by 80%
2. Cut characterization time to 6 months
3. Characterize propellant at pressure range of 200 -1000 psi.
4. Function for 10 years
5. Withstand three times max operation pressure
6. Calculate Burn rate to 5% accuracy
7. Calculate 'a' and 'n' values to 8% accuracy





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# Design Solution



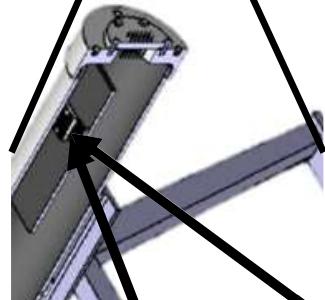
# System Overview

Pressure System

**Strand Burner:** a pressurized chamber used to characterize solid propellants



Propellant Strand vs French Fry



[1]

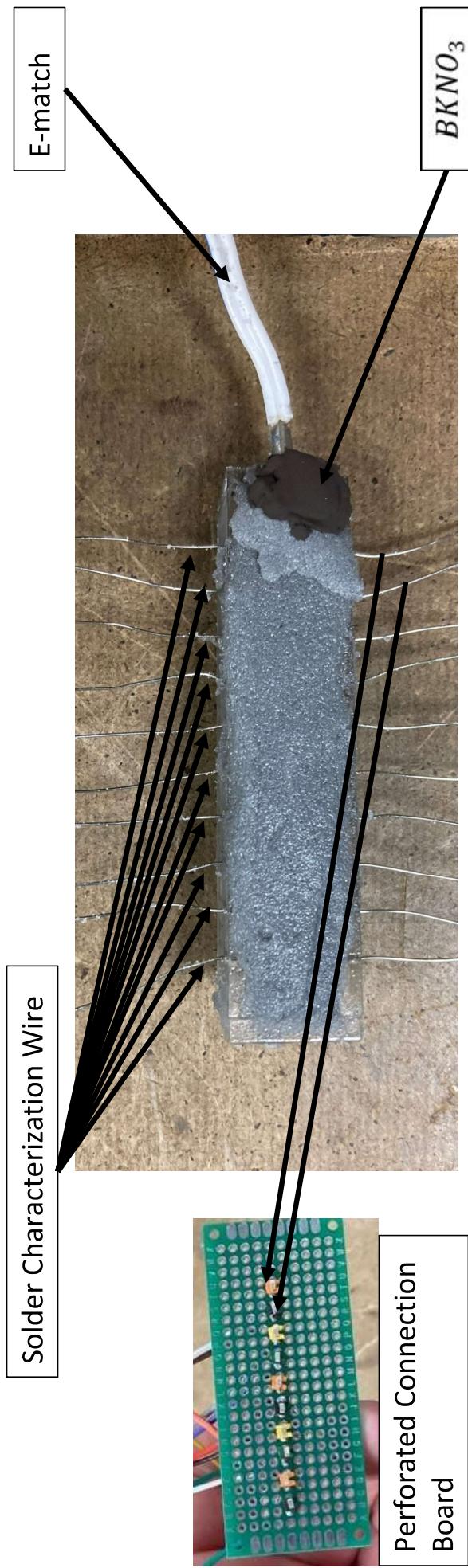
Structures  
Ignition

Data Acquisition and Software Interfacing

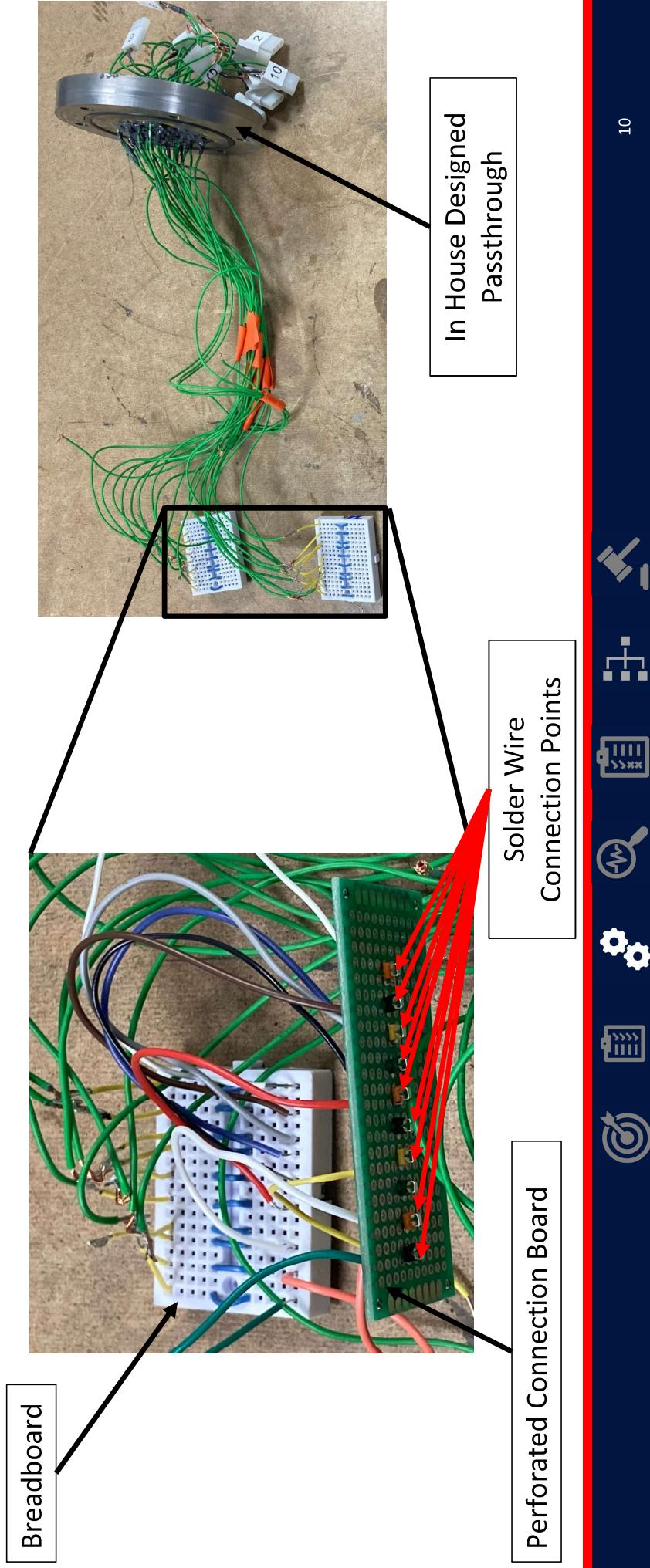
$$\text{Saint Roberts Law } \dot{r} = aP_c^n$$



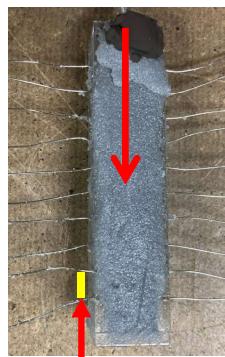
# System Overview



# System Overview



# Linear Burn Rate Equation and Saint Roberts Law



$W_d$ : distance between two wires  
Measured using calipers  $\pm .05$  mm

$\dot{r}$ : burn rate of the propellant and input to Equation Two

$n$ : pressure exponent derived from curve fit

$a$ : burn rate coefficient derived from curve fit

$$\dot{r} = \frac{W_d}{t} \quad [3]$$

$\dot{r}$ : burn rate and input from Equation One

$P_c$ : chamber pressure measured from pressure transducer to  $\pm 7$  psi.

$t$ : time to burn the wires measured using PXIe  $\pm 0.001$  s

$$\dot{r} = a P_c^n \quad [2]$$

# Materials and Construction – Strand Burner

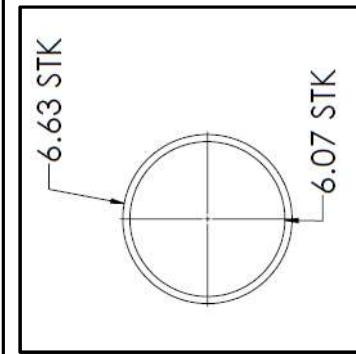
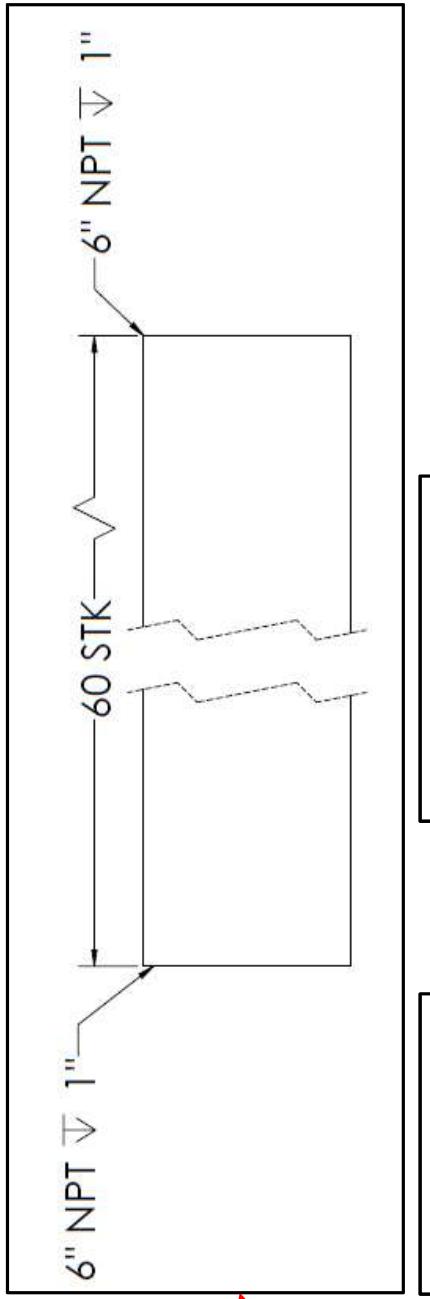


ASTM A500 Grade B threaded steel pipe  
Pipe threading outsourced

12L14 Cold rolled steel face plate  
Machined in house

1018 Hot roll steel threaded cap  
Machined in house

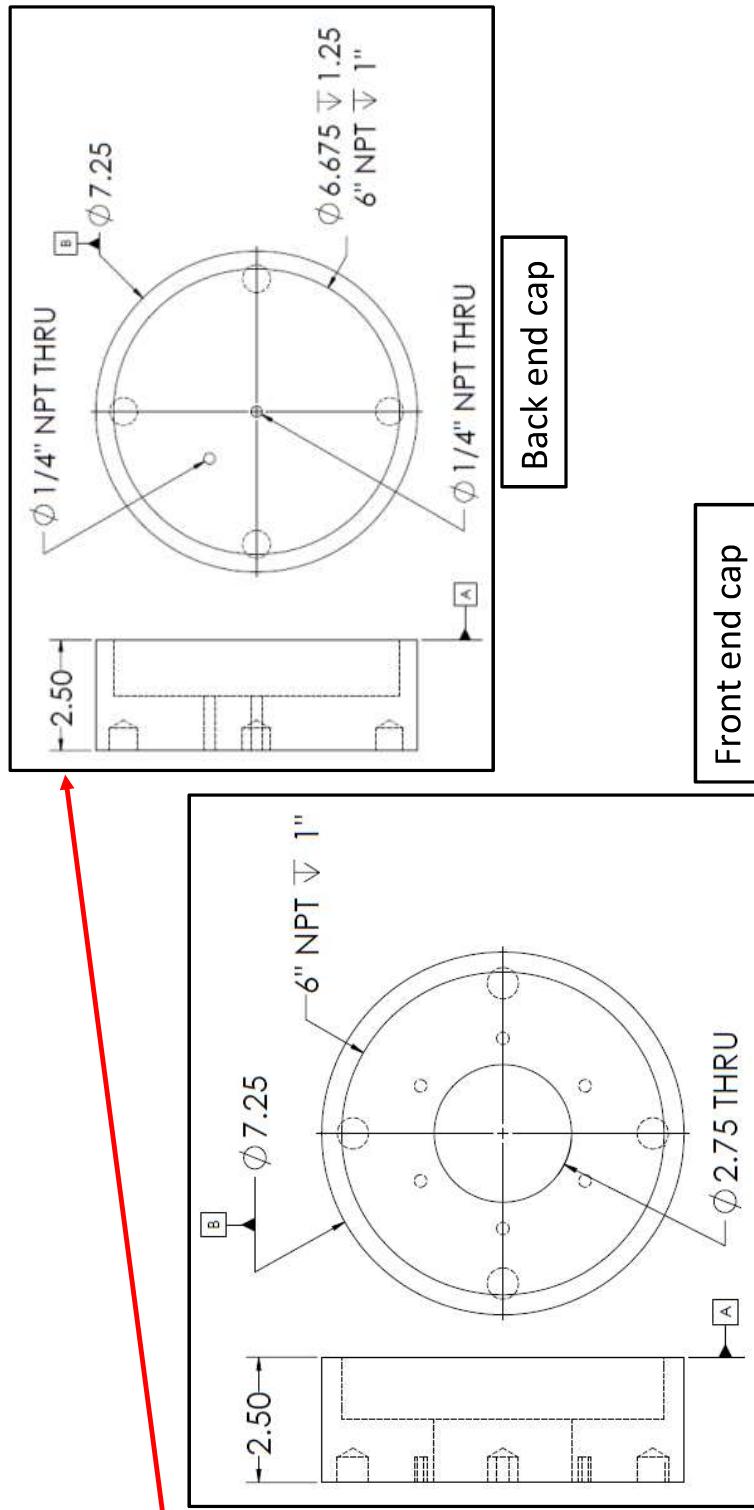
# Dimensions – Strand Burner



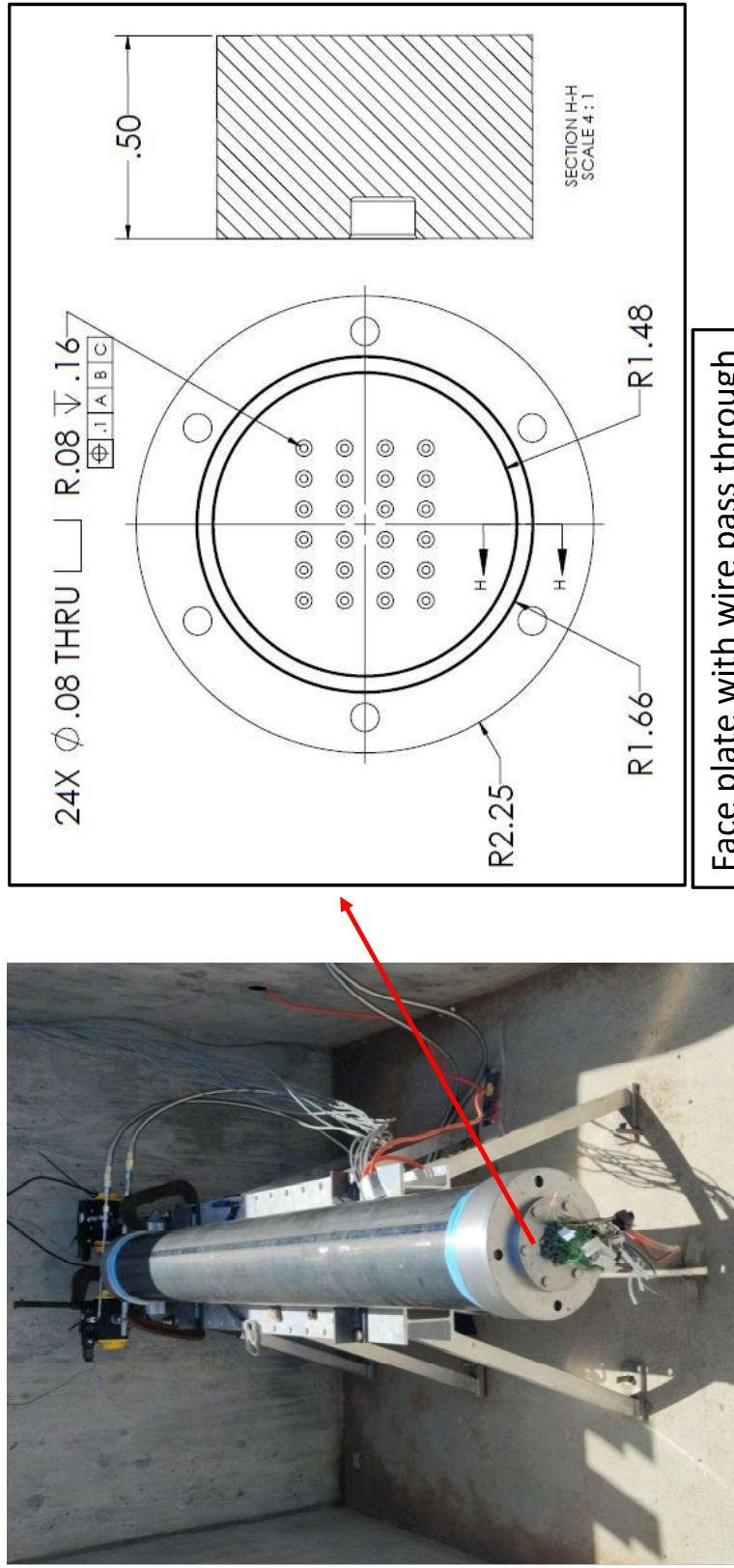
Pipe drawing



# Dimensions – Strand Burner Cont.



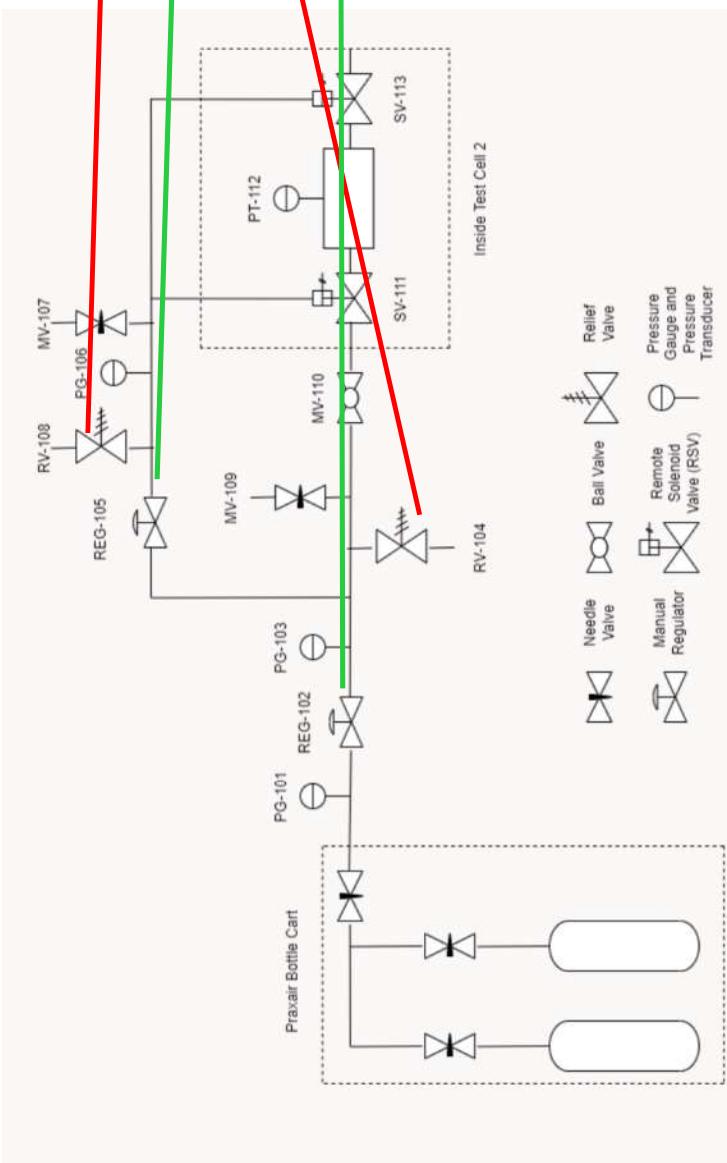
# Dimensions – Strand Burner Cont.



Face plate with wire pass through



# Materials and Construction – Pressure System



Back of pressure panel

# Materials and Construction – Pressure System



Flex hoses used to take gas  
to and from panel

304 stainless steel tubing;  $\frac{1}{4}$   
in. OD; 0.28 in. wall thickness

Steel AN and NPT  
pressure fittings



8020 rails used for panel frame and cut  
to desired length

Panel is cut from aluminum composite sandwich

Pre-assembly of the pressure components  
Cut holes according to the components'  
positions and assembled the panel



# Materials and Construction – Propellant Strand

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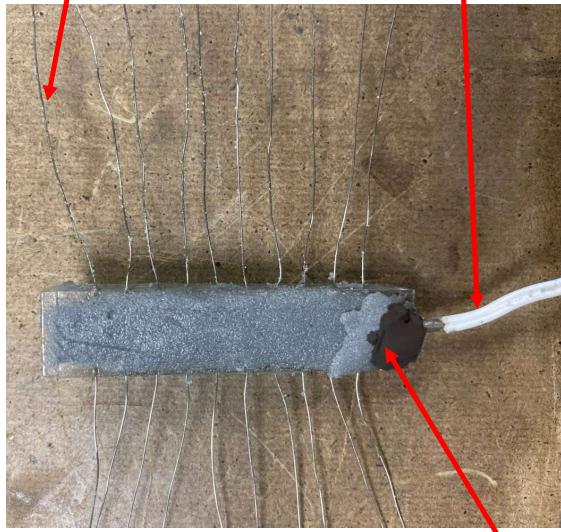


Propellant mixing was  
done by hand



Propellant jig made from plexiglass  
Sides are laser cut and glued together

# Dimensions and Construction – Propellant Strand



0.3 mm diameter lead free solder wire  
10 wires inserted at equal intervals with 0.143 in.  $\pm$  0.01 spacing

E-match

2 in. x 0.5 in x 0.5 in.

Propellant packed into the jigs, the wires and an E-match were inserted  
After curing, the BKNO<sub>3</sub> was added



# Materials and Construction – Data Acquisition and Software Interfacing

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NEMA-4 rated box holds  
the connection for wires  
going to the PXle



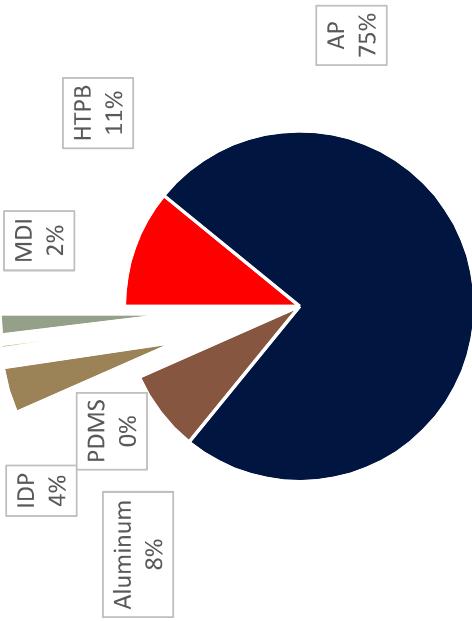
PXle [13]



# Supporting Analyses



# Performance Analysis



■ HTPB ■ AP ■ Aluminum ■ IDP ■ PDMS ■ MDI

Below melting point of  
structural components

## Simulation Results from NASA

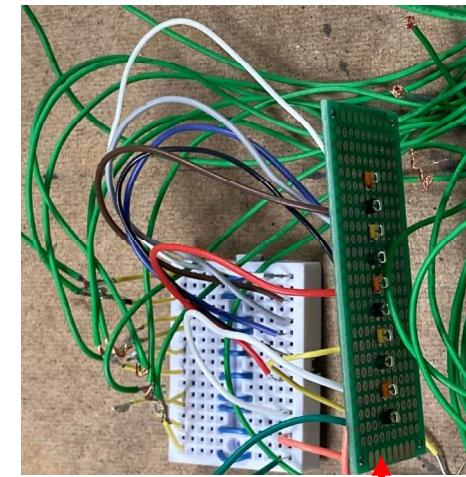
Properties	Values
Gas Temperature, $T_g$	306 K, 91°F
Energy Released, $\Delta U$	797.45 kJ/kg
Specific Heat Ratio, $\gamma$	1.227
Molecular weight, M	27.356 g/mol



# Performance Analysis

- Ideal Gas Law → Pressure Rise
- Gas temperature below melting point of all components

Copper-coated Fiberglass perf boards shield heat locally



# Structural Analysis – Pipe Thickness

From ASME Pressure Vessel Codes<sup>[7]</sup>

$$th = \frac{r_i P}{A_s E - 0.6P}$$

Minimum thickness of pipe  
Inner radius from Pressure Analysis  
Chamber Pressure of 3000 psi  
Additional thickness due to threads

Max. allowable stress of material  
Joint Efficiency  
Threads per inch

$$t = \frac{0.8}{n}$$



# Structural Analysis – Pipe Thickness

From ASME Pressure Vessel Codes<sup>[7]</sup>

$$th = \frac{3.0125 * 3000}{* 1 - 0.6 * 3000}$$

Minimum thickness of pipe  
Inner radius from Pressure Analysis  
Chamber Pressure of 3000 psi  
Additional thickness due to threads

Max. allowable stress of material  
Joint Efficiency  
Threads per inch

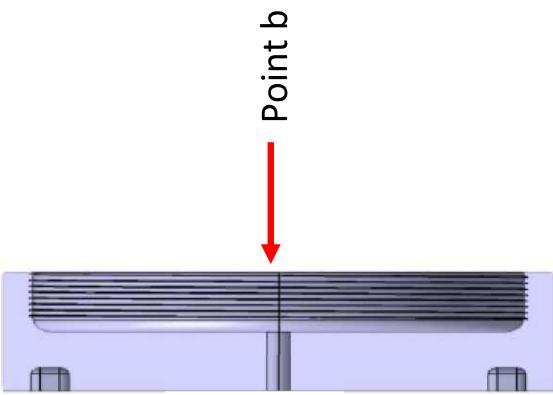
$$t = \frac{0.8}{n}$$



# Structural Analysis – Threaded Cap Thickness

Moment at point b  
[8]

Stress at point b [8]



$$M_{rb} = qa^2 \left( \frac{L14}{C5} \right)$$

$$\sigma_{rb} = \frac{6M_{rb}}{th^2}$$

$M_{rb}$  = Moment at point b

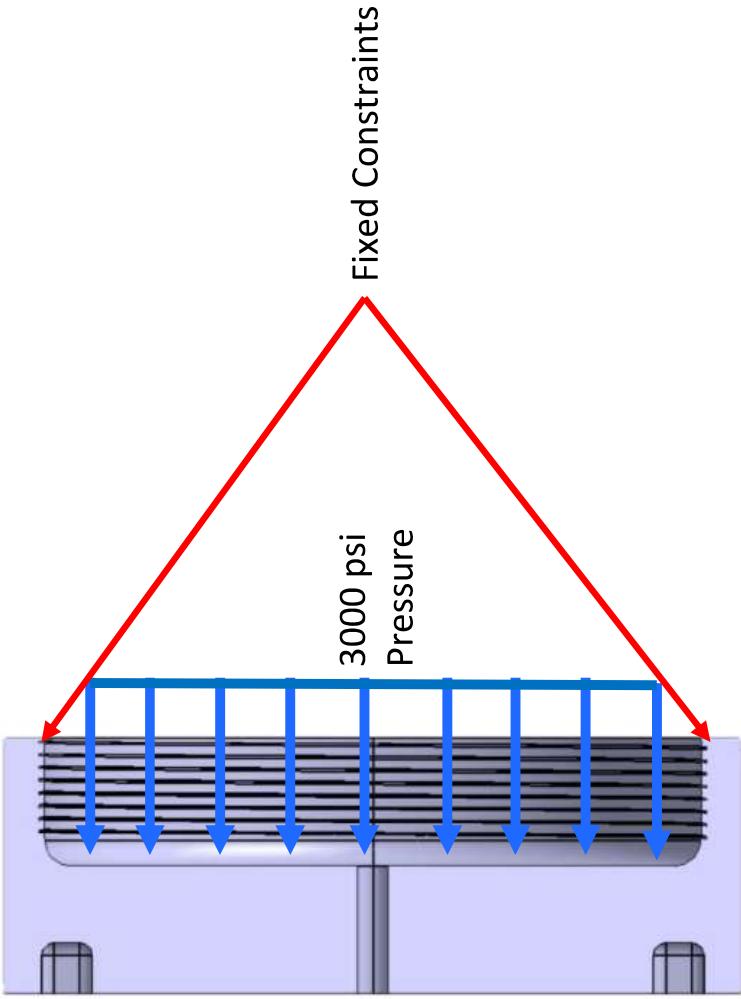
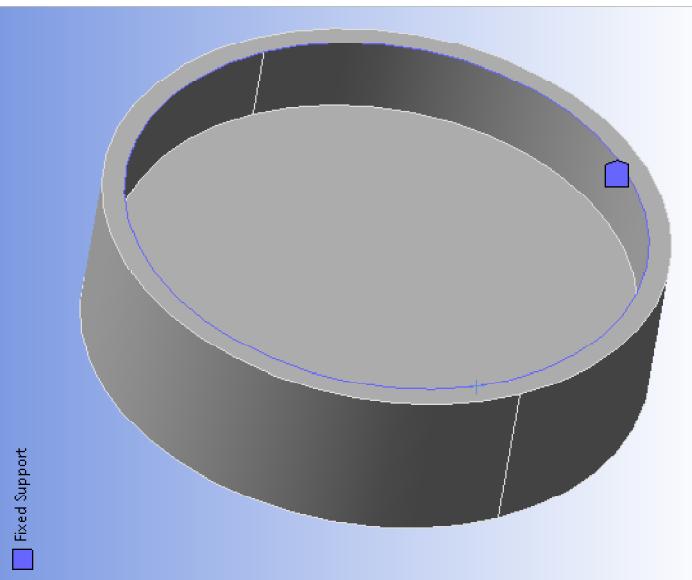
q = Pressure

a = Radius of plate

L14 & C5 = Constants

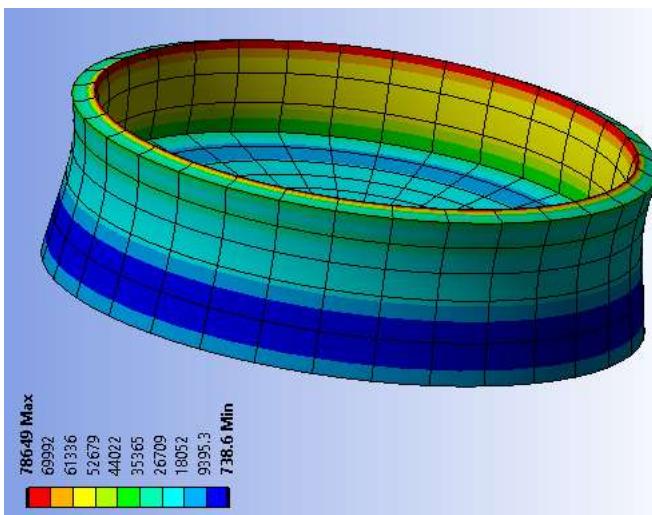
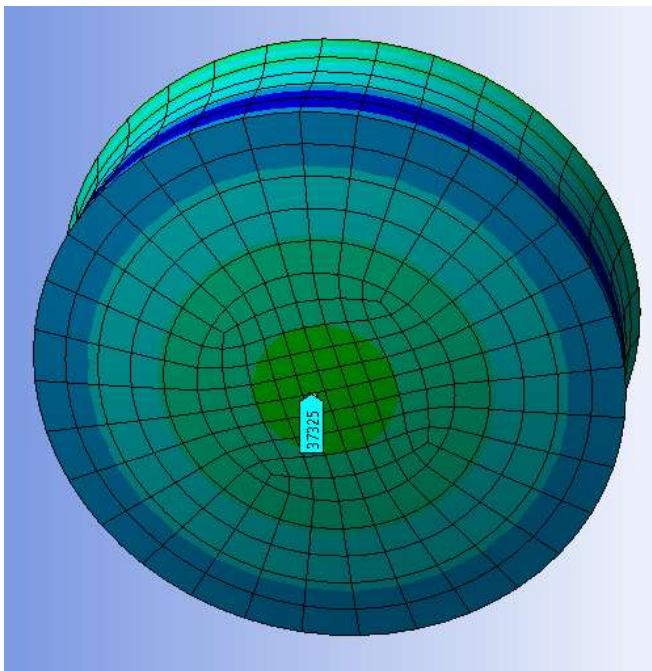
$\sigma_{rb}$  = Stress at point b  
th = Minimum thickness of caps

# Structural Analysis – Numerical Analysis of Threaded Caps



# Structural Analysis – Numerical Analysis of Cap Thickness

- Maximum stress: 37325 psi
- Hand calcs: 38352 psi
- Tensile strength: 58000 psi



# Structural Analysis – Caps and Bolts

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Number of threads in caps [9]

$$n = \frac{F_p}{0.5\tau_s p \pi d}$$

$$\sigma = \frac{\pi d^2 p}{\text{_____}}$$

2863 psi

d = Diameter of face plate

p = Chamber pressure

$\sigma$  = Stress in bolts

Tensile strength of  $\frac{1}{4}$  28 bolts:

3000psi

n = Number of threads

$\tau_s$  = Shear strength

p = Pitch

d = Pitch diameter

$F_p$  = Force on the plate



# Structural Analysis – Fatigue Analysis

Number of cycles before failure [10]

$$a = \frac{(fS_{ut})^2}{S_e}$$

$$b = -\frac{1}{3} \log \left( \frac{fS_{ut}}{S_e} \right)$$

$$N = \left( \frac{\sigma_{rev}}{a} \right)^{\frac{1}{b}}$$

f = Fatigue strength fraction

$S_{ut}$  = Ultimate tensile strength

$S_e$  = Endurance limit

a = Constant

b = Constant

N = Number of cycles before failure

$\sigma_{rev}$  = Completely reversed stress

$$\boxed{N = 37430 \text{ cycles}}$$



# Uncertainty Analysis

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Burn Rate Calculation:

$$U_{W_d} = \pm 0.001 \text{ in}$$

$$U_t = \pm 0.001 \text{ seconds}$$

$$UU_{\dot{r}} = \dot{r} \sqrt{\left(\frac{U_{W_d}}{W_d}\right)^2 + \left(-\frac{U_t}{t}\right)^2}$$

$$\circ = \pm 0.001 \text{ in/s}$$

'a' and 'n' value Calculation

$$\bullet \text{ Assuming } \frac{U_a}{a} = \frac{U_n}{n}$$

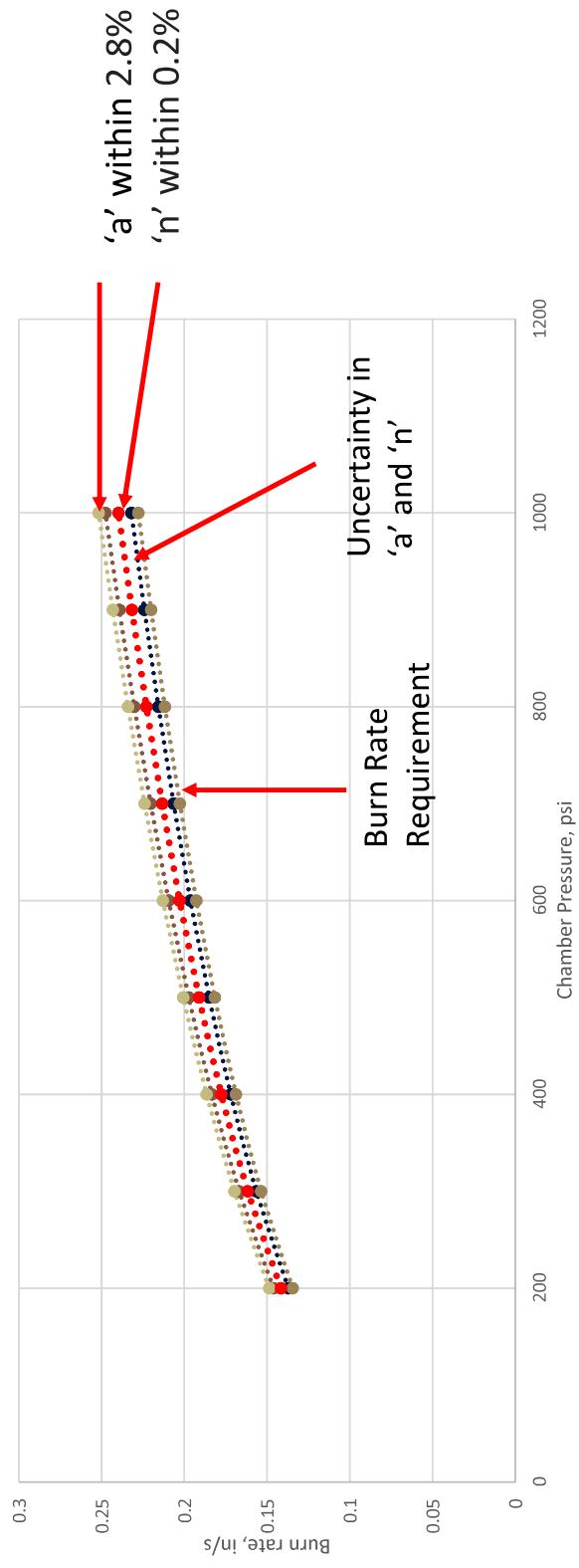
$$\bullet U_P = 7.5 \text{ psi}$$

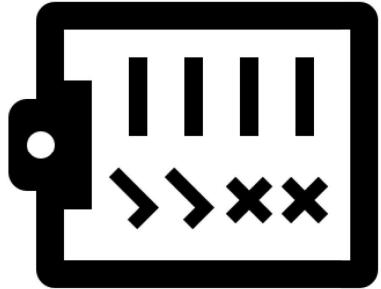
$$\bullet U_r \text{ total} = 0.001 \frac{\text{in}}{\text{s}}$$

$$\bullet U_a = \sqrt{\left(\frac{\partial a}{\partial \dot{r}} U_{\dot{r}}\right)^2 + \left(\frac{\partial a}{\partial P} U_P\right)^2 + \left(\frac{\partial a}{\partial n} U_n\right)^2}$$

$$= \pm 0.0007 \text{ in/s}$$

# Uncertainty Analysis



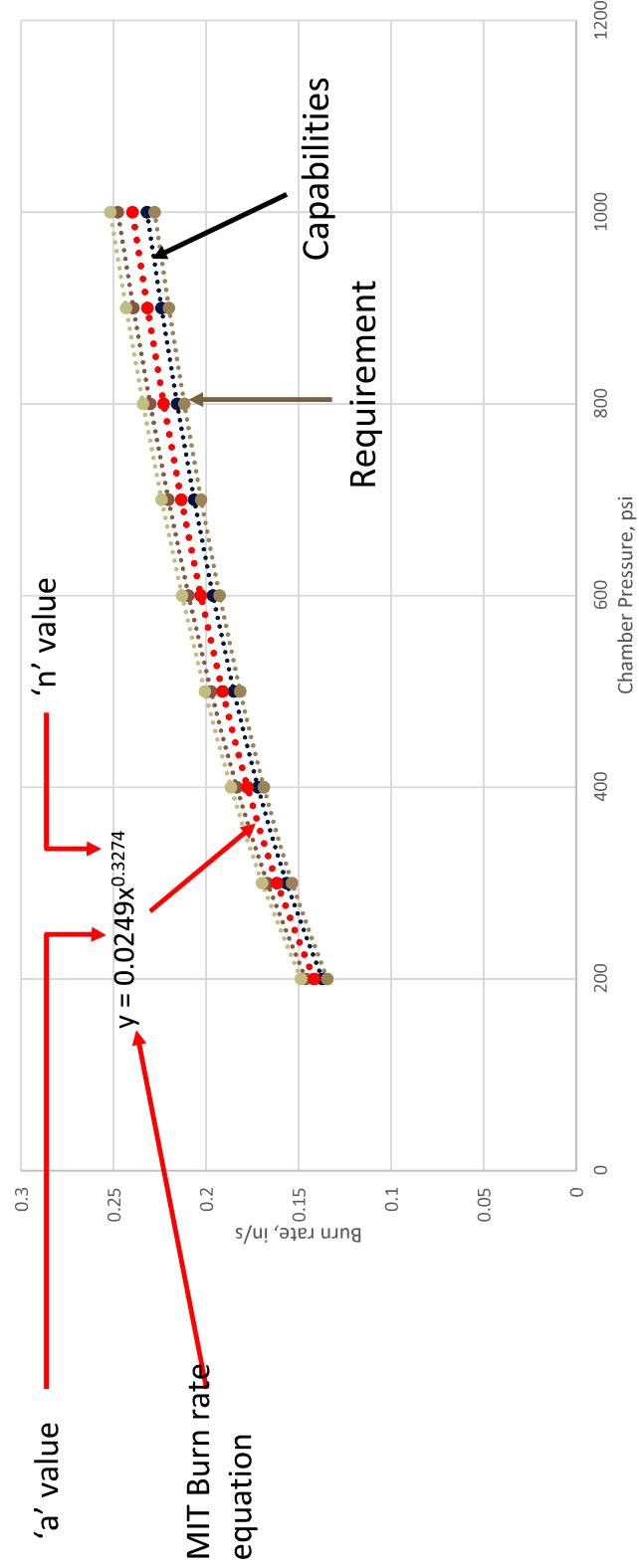


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# Verification Tests



# Projected Outcomes Derived From MIT Data



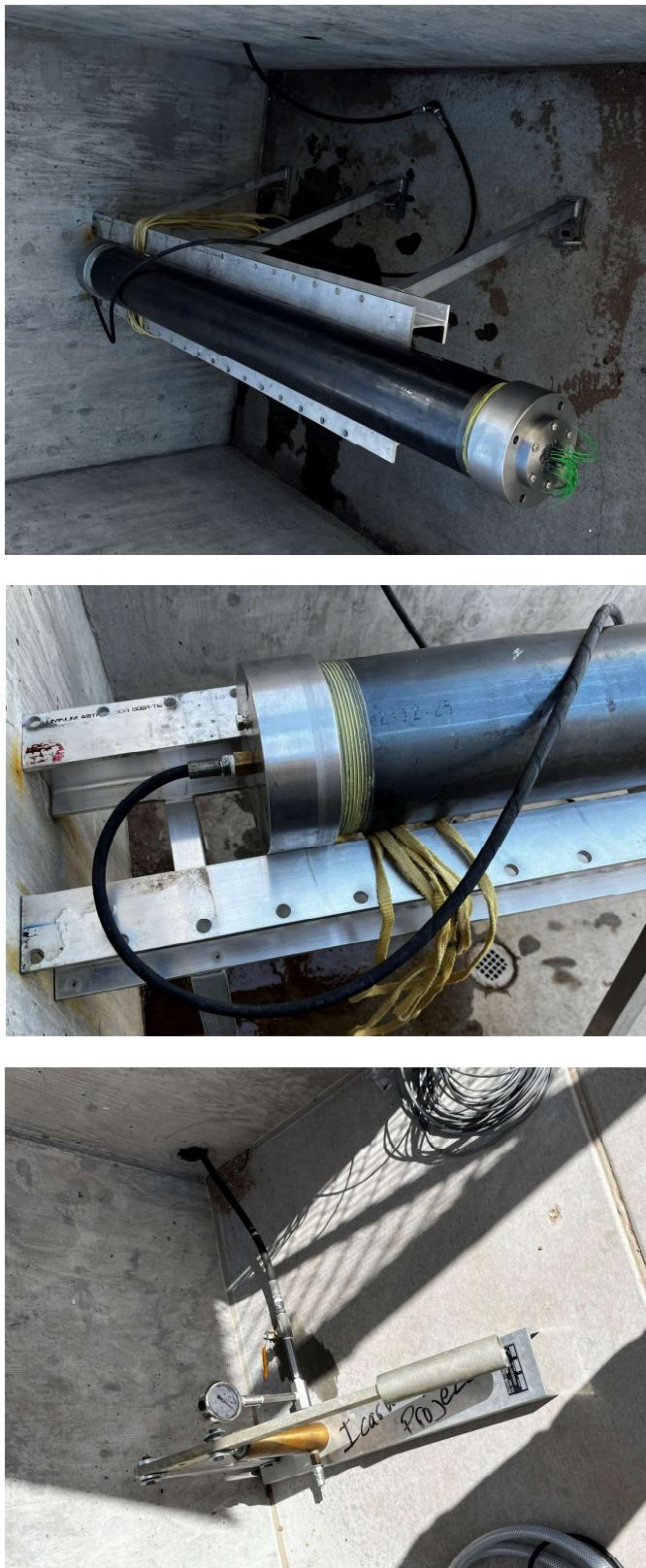
# Verification Plan

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# Hydrostatic Test

Hydrostatic Test Setup at Test Cell 2



# Hydrostatic Test

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- Result:

- 3 Attempts
- Failed
- Max pressure – 1600 psi.



Test 2



Test 3



# Propellant Burn Test

- BKNO<sub>3</sub> Igniter Worked
- Wires Burned
- Kept Jig On
- Passed



During Ignition Test

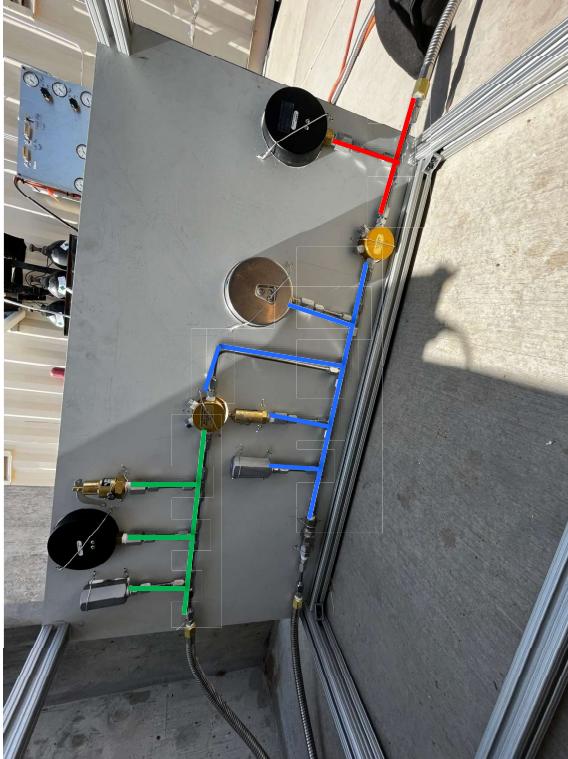


After Ignition Test



# Leak Check of Panel

- 1600 psi. in Bottle
- 100 psi. Check
- Check Red, Blue, Green
- **Passed**



Sections of Pressure Panel and Strand Burner



# Click Check

- 100 psi. Supply
- Strand Burner Press Valve  
**Failed**



Setup of Actuators



Actuator on Valve



# Leak Check of Strand Burner

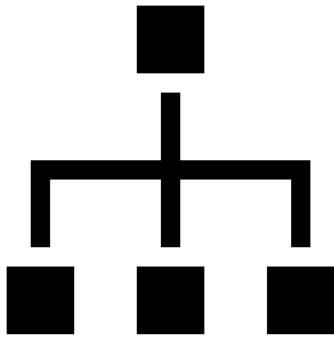
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- 1600 psi. in Bottle
- 100 psi. Check
- Failed
- Could not Continue to Characterization Campaign



Strand Burner Setup for Leak Check





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# Compliance Summary

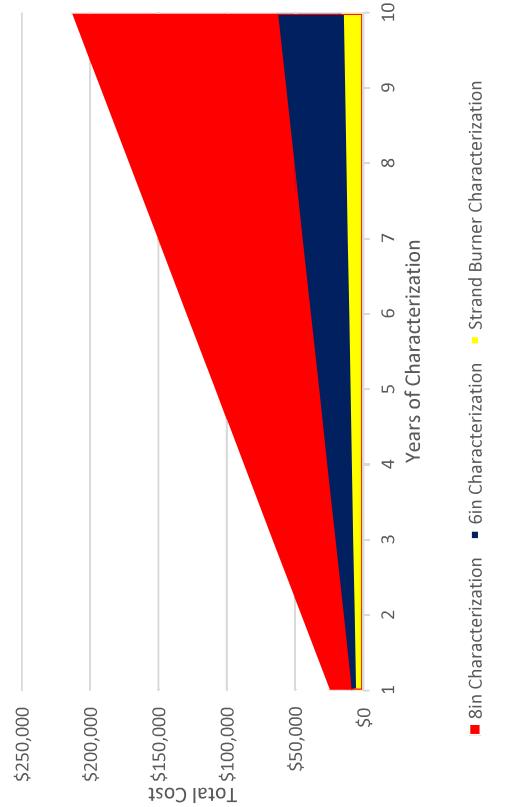


# Requirement 2.1 - Compliance

Project must cut cost of characterization by 80%

Status: COMPLETE

Characterization Cost Comparison



■ 8in Characterization ■ 6in Characterization ■ Strand Burner Characterization

Method	Initial Cost	Reoccurring Cost	Total Cost at 10yrs
8in Characterization	\$4,000	\$21,000	\$214,000
6in Characterization	\$4,000	\$6,000	\$64,000
Strand Characterization	\$6,000	\$1,000	\$16,000

Initial Costs: Chambers, test stands, reusable equipment

Reoccurring Costs: Chemicals



# Requirement 2.4 - Compliance

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System must function properly for 10 years.

Status: COMPLETE

$$N = \left( \frac{\sigma_{rev}}{a} \right)^{\frac{1}{b}}$$

Rubber Coating



Rubberized coating application to chamber interior

N = 37430 cycles



# Requirement 2.2 - Compliance

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Lower time of characterization to 6 months for all rocket motor sizes used by ERAU

Status: FAILED

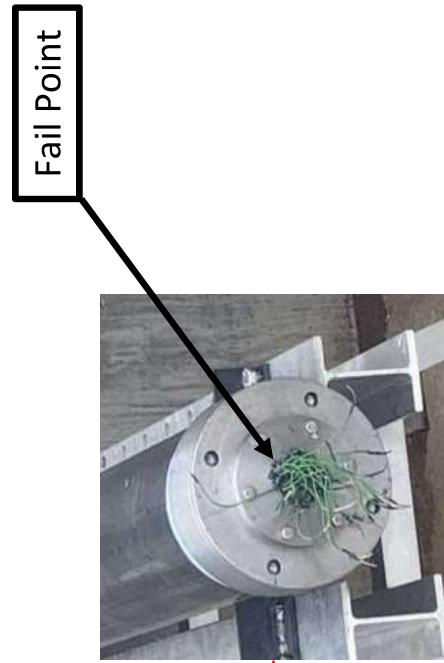


# Requirement 2.3 - Compliance

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System must be able to characterize propellant at chamber pressure of 200 - 1,000psi.

Status: FAILED



# Requirement - Compliance

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2.5

System must withstand up to 3 times the max operating pressure without failure

Status: FAILED



2.6

Calculate Burn rate to 5% accuracy

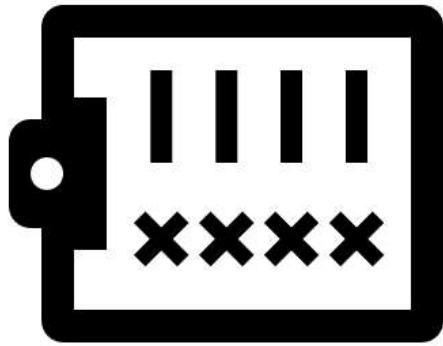
Status: INCOMPLETE

2.7

Calculate 'a' and 'n' values to 8% accuracy

Status: INCOMPLETE





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# Failure Analysis



# Hydrostatic Test

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- Failure Modes:

- Liquid Thread Seal – Replaced with Teflon
- Wire Pass Through
- Max pressure – 1600 psi.



Test 2



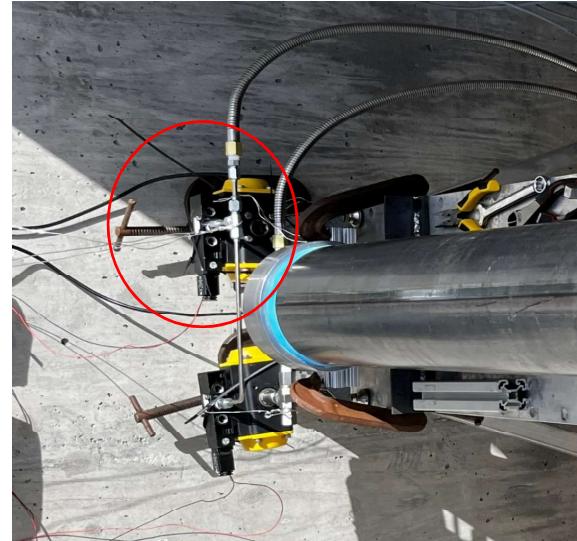
Test 3



# Remote Valve Actuators

## Failure Mode:

- Thought it was Alignment
- Strand Burner Press Valve  
Actuator Broken



Setup of Actuators



Actuator on Valve



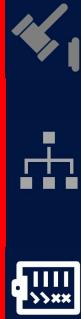
# Leak Check of Strand Burner

## Failure Mode:

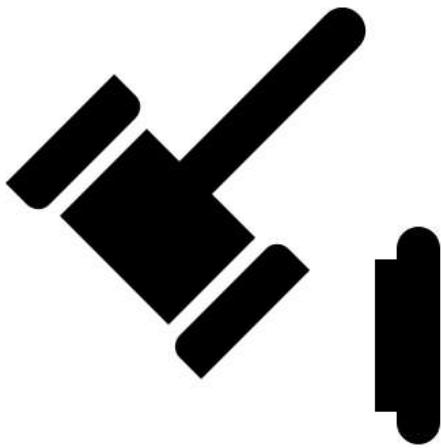
- ~100 psi.
- Snapped Caps, Faceplate,  
Plumbing – No Leaks
- Pass Through Whistling –  
Determined to be Leak Point



Strand Burner Setup for Leak Check



# Conclusion and Recommendations



# Conclusion

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## Project accomplishments:

- Theoretically reduces the cost of characterization up to 80%
- Test article life span up to 10 years

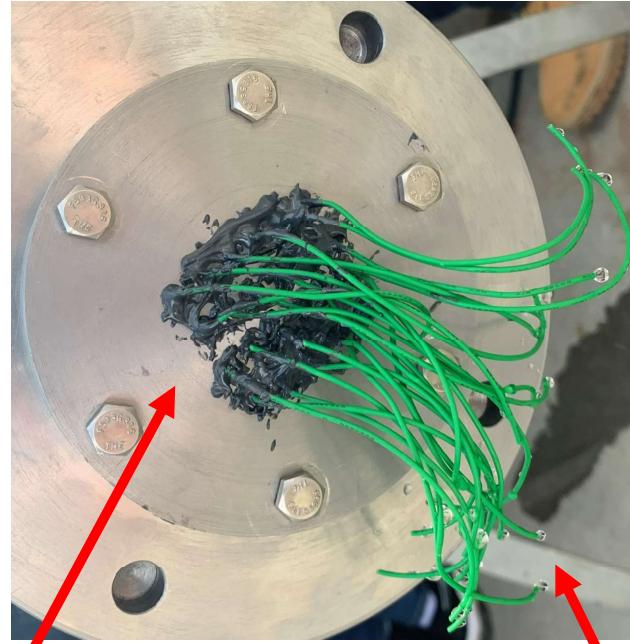
## Project setbacks:

- Supply chain delays
  - Wire pass through
  - Argon
  - Hydrostatic pump



# Project Failures

- Face Plate wire pass through
- Hydrostatic Test (Sealing)
- Actuator



Wire Pass Through

Water Droplets

# Recommendations

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## Buy Commercial Passthrough

- Screws into plate via NPT fitting
- Connections are more reliable



## Make Passthrough



- Solid core Wires
- Better labeling
- New way to seal wire holes
- Strip and seal wires
- Shorter wires

# Recommendations Cont.

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- **Perform full hydrostatic test**

- Make a test plate
- Use new pump
- New method of filling strand burner



- **Seal the inside of the caps with rubber**

- Prevent rust
- Use blue Teflon for threads



# Recommendations Cont.

## Pressure/Data Acquisition system:

- New actuator
- Attachable valve harness to support the valves
- LabVIEW code
  - Run all LabVIEW through PXIe to control actuation and data acquisition



Actuator Support



Actuator





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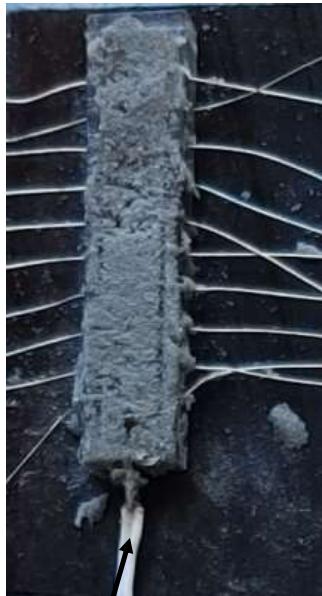
## Future Improvements



# Future improvements

## Propellant sample:

- Redesign propellant jig
- Ignition method E-match
- BKNO<sub>3</sub> (boron/potassium nitrate)
- Mix enough propellant for a full-size motor to verify test results



# Future Improvements Cont.

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## Future Addons:

- Patch Box Capabilities
  - Analog
  - Digital
- Bridge Completion
- Color Coding
- Thermocouple
- Integrate the liquids rocket control box to control the ignition system and actuation
- Propellant combustion byproduct collection system



# Acknowledgments

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- Dr. Steven F. Son, Purdue University
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- Dr. Elliott Bryner, Assistant Professor of Mechanical Engineering
- Dr. Daniel White, Assistant Professor of Mechanical Engineering
- Dr. Brent Solie, Associate Professor of Mathematics
- Dr. Karl Heine, Assistant Professor of Mechanical Engineering
- Dr. Katherine Wood, Associate Professor of Engineering
- Dr. Jonathon Adams, Assistant Professor of Humanities and Communication
- Professor Robert Erickson, Department Chair of Engineering
- Jeffrey Hiatt, Engineering Lab Specialist
- Jared Anatta, College of Engineering Lab Manager
- Patrick David,achinist
- Andrew Hamilton, mechanical Engineering Student
- McNeil W. Allison, Industrial Engineer, Consultant – Global Healthcare Solutions
- Michael Brady, EHS Program Manager

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# Back-up Slides

## Dimensions – Strand Burner Cont.

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**Feet**  
2 12 in. x 2 in. x 2 in.

# System Overview

Pressure System [1]

Structures

**Strand Burner:** a pressurized chamber used to characterize solid propellants.

$$\text{Saint Roberts Law } \dot{r} = aP_c^n$$

**Goal:** Determine the 'a' and 'n' values

Ignition

Data Acquisition and Software Interfacing



# Linear Burn Rate Equation

---

$$\dot{r} = \frac{W_d}{t}$$

$\dot{r}$ : burn rate of the propellant and input to Equation Two

$W_d$ : distance between two wires  
Measured using calipers  $\pm .05$  mm

$t$ : time to burn the wires  
measured using PXIe  $\pm 0.001$  s

[3]



# Saint Roberts Law

$$\dot{r} = a P_c^n$$

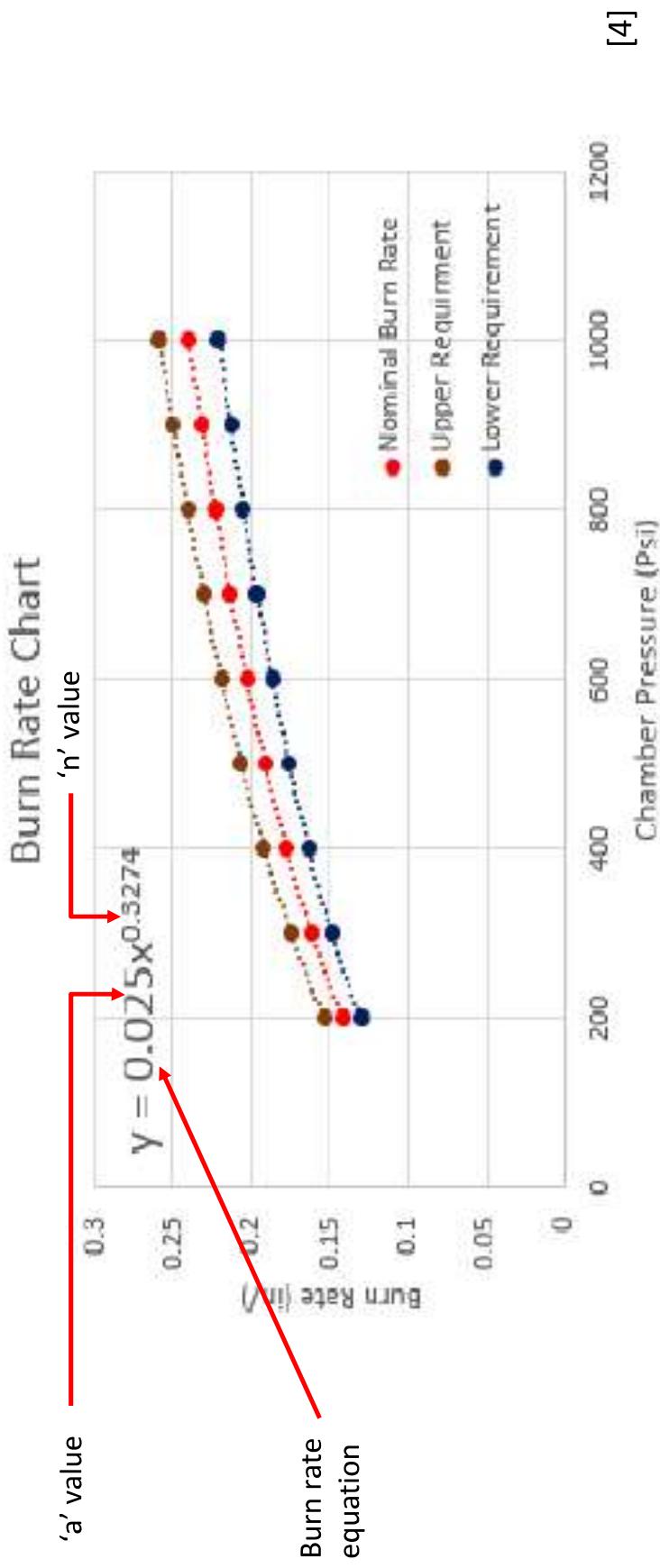
$P_c$ : chamber pressure measured from pressure Transducer to  $\pm 7$  psi.

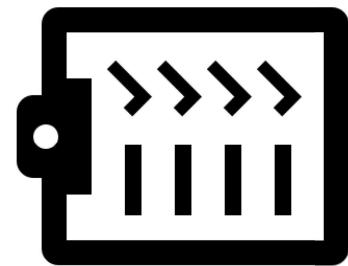
$a$ : burn rate coefficient derived from curve fit

$\dot{r}$ : burn rate and input from Equation One

$n$ : pressure exponent [2] derived from curve fit

# Projected Outcomes Derived From MIT Data





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# System Requirements



# System Requirements

---

# Test Objectives and Requirements

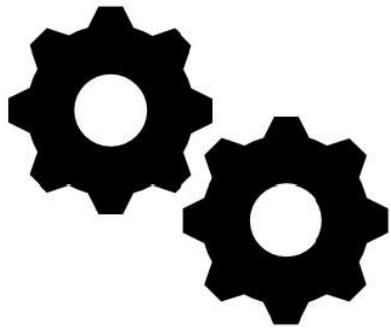
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## Objectives

1. Test accuracy of data collection methodology by comparing gathered data to MIT's posted values
2. Demonstrate the practicality of the strand burner
3. Accurately collect pressure data over the range of 200 - 1000 psi.
4. Accurately collect burn rate data over the range of 200 - 1000 psi.

## Requirements

- 1 Measure "a" to an accuracy of 8%
- 2 Measure "n" to an accuracy of 8%
- 3 Measure the burn rate accuracy to 5%
- 4 Pressure range must be 200 psi to 1000 psi
- 5 Testing shall take no longer than 6 months



---

# Test Article



# Strand Burner Chamber

- Holds test sample
  - Creates/maintains testing environment
  - Allows for instrumentation access to the test environment



Strand burner chamber ready to go

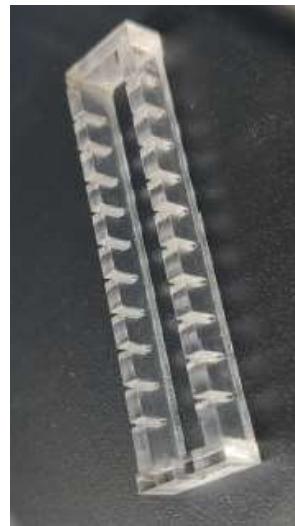


## Coating application to chamber

# Propellant Sample



Propellant Samples being made



Propellant Jig



Propellant Sample

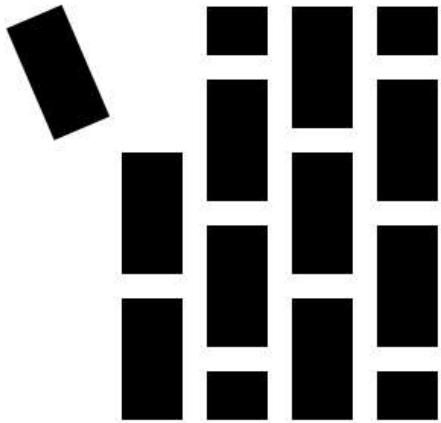


# Support System - Pressure System



Pressure Panel Current Construction Progress



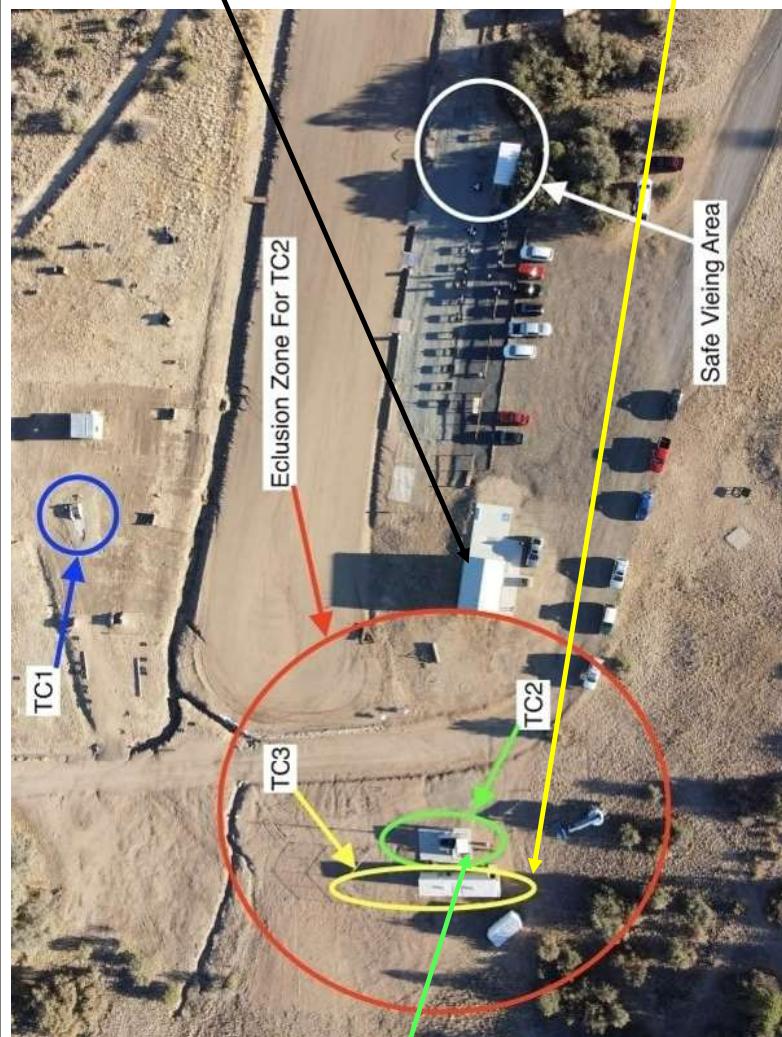


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# Test Facilities



# Test Facilities



Test Cell  
2

# Test Facilities—PXI Express (PXIe)

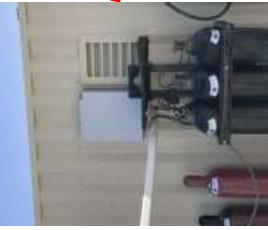
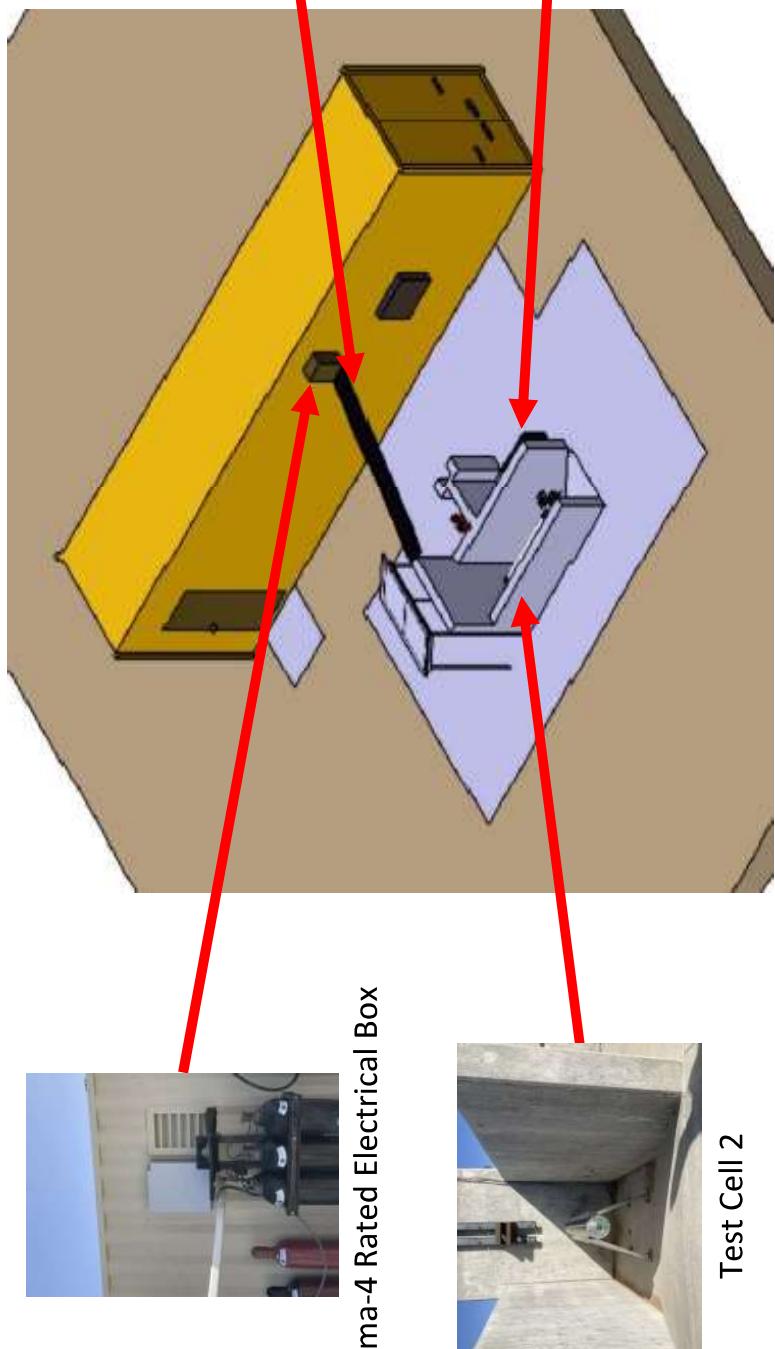
- Pressure Transducer Ports
- Voltage Source
- Voltage Block
- 20 Talk Back Channels
- Power Back Up
- Uncertainty:  $\pm 0.001$  sec



PXIe [13]



# Test Facilities—Integration



Nema-4 Rated Electrical Box



Catwalk



Test Cell 2



Pressure Panel





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# Instrumentation



# Instrumentation List

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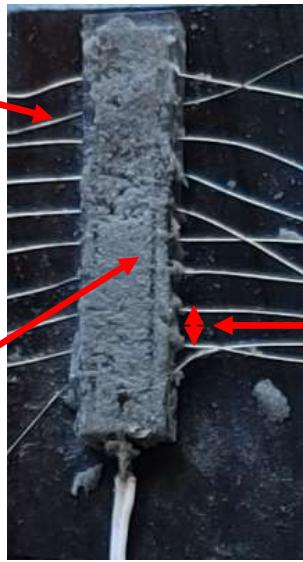
## Project Box and Resistors

- 9 Resistors in parallel inside Project Box
- Time of voltage drop across resistors recorded using PXle to  $\pm 0.001$  seconds
  - Under construction
- 3000 psi Pressure Transducer [9]
  - Measures chamber pressure to  $\pm 7$  psi
  - Response time < 1 millisecond
  - Can measure up to 3000 psi



# Instrumentation List Cont.

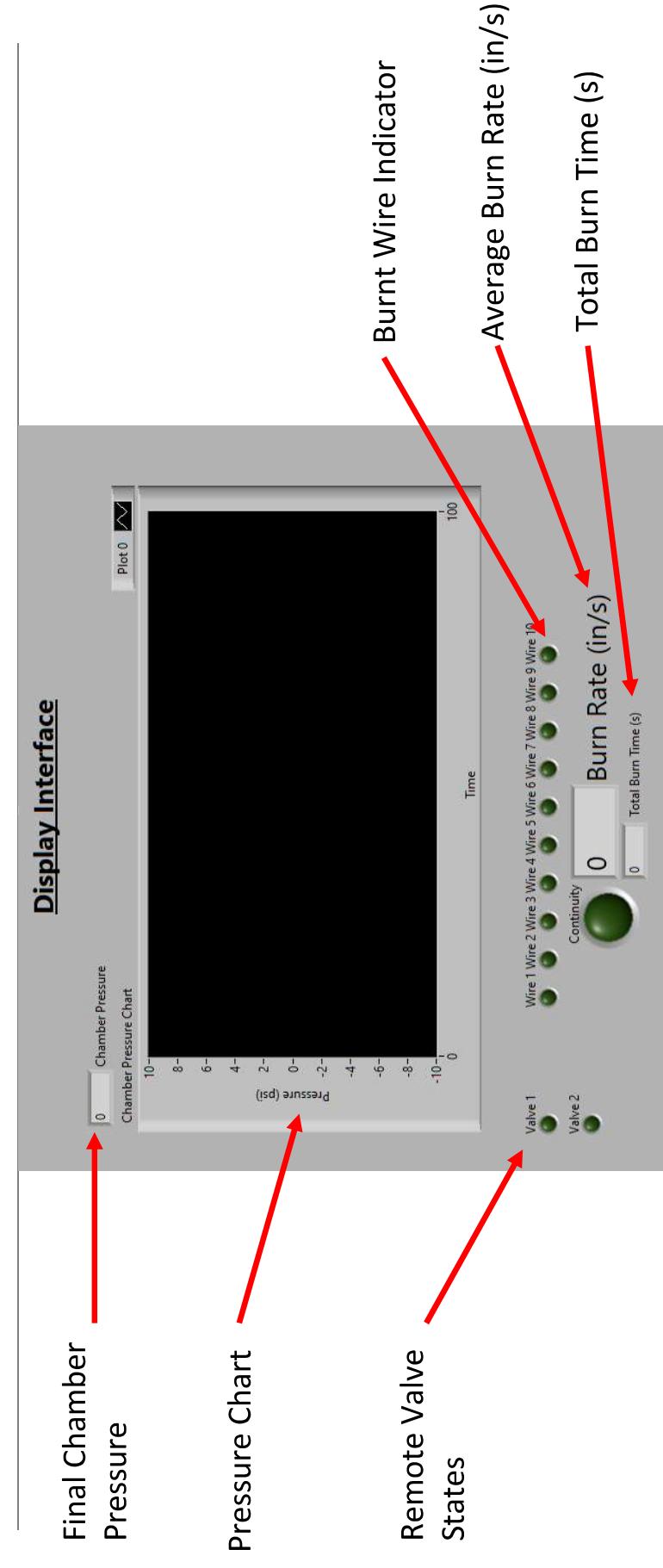
Soldering wire  
Propellant strand



- Standardizes distance between wires across several strands
  - The wire slots measured to  $\pm 0.001$  in

Measured before the test using  
Calipers

# Instrumentation List Cont.



# Uncertainty Analysis

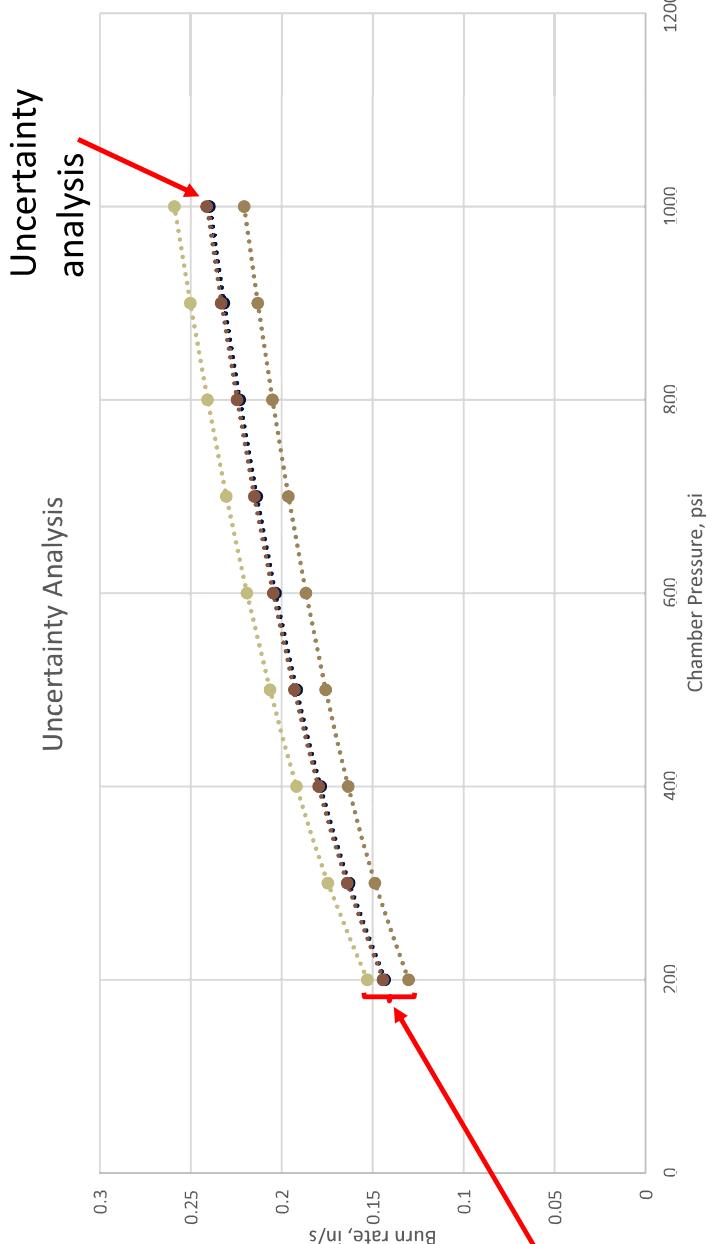
Burn Rate Calculation:

$$U_{W_d} = \pm 0.0005 \text{ in}$$

$$U_t = \pm 0.001 \text{ seconds}$$

$$UU_r = \dot{r} \sqrt{\left(\frac{U_{W_d}}{W_d}\right)^2 + \left(-\frac{U_t}{t}\right)^2}$$

$$\circ = \pm 0.001 \text{ in/s}$$



Required upper and  
lower limits

# Uncertainty Analysis Cont.

---

'a' and 'n' value Calculation

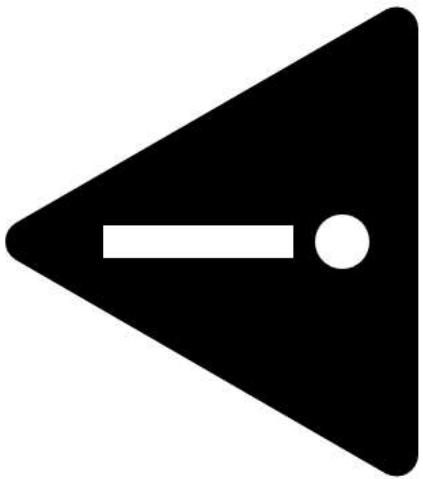
- Assuming  $U_a = U_n$

- $U_P = 7.5 \text{ psi}$

- $U_r = 0.001 \frac{\text{in}}{\text{s}}$

$$U_a = 0.0003 \text{ in/s}$$

$$\bullet U_a = \sqrt{\left(\frac{\partial a}{\partial r} U_r\right)^2 + \left(\frac{\partial a}{\partial P} U_P\right)^2 + \left(\frac{\partial a}{\partial n} U_n\right)^2}$$



---

# Risk Analysis



# Risk to Personnel – Risk Cube

		Risk Cube: Personnel				
Risk Probability/ Severity	Catastrophic (A)	Critical (B)	Moderate (C)	Minor (D)	Negligible (E)	
5			35			
4			2, 1, 5, 6, 14, 27			
3				13, 22		
2		3	21	12, 32		
1	33, 34	4, 19	20			

		Trippling/Slipping/Falling/Health Hazards				
Risk Probability/ Severity	Catastrophic (A)	Critical (B)	Moderate (C)	Minor (D)	Negligible (E)	
5					• 1-6, 32, 35	
4					• 13-14, 20-22, 27	
3					• 19, 34	
2					• 33	
1						

**Handling of Test Article/Equipment**

- 13-14, 20-22, 27

**Rapid Unscheduled Disassembly**

- 19, 34

**Argon Exposure**

- 33

- A) multiple deaths  
 B) one death; serious/multiple long-term injuries or resulting in hospitalizations  
 C) minor injury  
 D) no injury or minor injury  
 E) no injury



# Risk to Personnel – Mitigations Cube

Mitigation Cube: Personnel						Markings/Signage In and Around Test
Risk Probability/ Severity	Catastrophic (A)	Critical (B)	Moderate (C)	Minor (D)	Negligible (E)	Cell 2
5						• 2-3, 6 Proper Training • 12, 13, 22, 27
4						Handling of Test Article/Equipment • Use of X-frame • 14, 21, 33-34
3						Personnel Limitation/Exclusion Zone • 1, 19-20
2						Other • 4-5, 32, 35
1				1, 2, 3, 4, 6	12, 13, 14, 20, 27, 22	5, 19, 21, 32, 33, 34, 35



# Risk to Facilities and Equipment – Risk Cube

Risk Cube: Facilities and Equipment

Risk Probability/ Severity	Catastro phic (A)	Critical (B)	Moderat e (C)	Minor (D)	Negligible (E)
5					
4		30			10
3		17, 18, 23, 37, 38, 43	15, 24, 25, 28, 29, 36		
2		16, 26		44	
1	34		11		

Installation/Handling of Test Article/Equipment

- Operations
  - 15-16, 44
  - 10, 36-38, 43
  - 11
- Manufacturing
  - 17-18, 23-26, 34
  - 28-30
- Rapid Unscheduled Disassembly
  - 17-18, 23-26, 34

- A) cost of \$250 or more or complete destruction of TC2 and/or equipment  
 B) \$50k or more; severe damage to equipment  
 C) \$5000 or less; some damage to equipment
- D) \$1000 or less; slight to no damage to equipment  
 E) \$100 or less; no damage to equipment



# Risk to Facilities and Equipment – Mitigations Cube

Mitigation Cube: Facilities and Equipment						Inspection of Setup
Risk Probability/ Severity	Catastrophic (A)	Critical (B)	Moderate (C)	Minor (D)	Negligible (E)	Operations
5						• 11, 16 • 16, 26, 34
4						Handling of Test Article/Equipment • 15, 44
3						Design Parameters • 10, 17-18, 23-25, 36-38, 43
2						Storage of Test Article/Equipment • 28-30
1						



# Risk to Success of the Test – Risk Cube

Risk Cube: Success of the Test					
Risk Probability/ Severity	Catastrophic (A)	Critical (B)	Moderate (C)	Minor (D)	Negligible (E)
5				31	
4		30	7		
3		17, 18, 23, 24, 25, 43	15, 28, 29, 36, 37, 38, 42	39, 40, 41	
2			16, 26	8, 9	12, 44
1		34		11	

- A) greater than a month schedule delay;  
 B) schedule delay greater than two weeks  
 C) schedule delay greater than three days

- D) schedule delay greater than 1 day  
 E) schedule delay due to equipment malfunction (couple of hours)

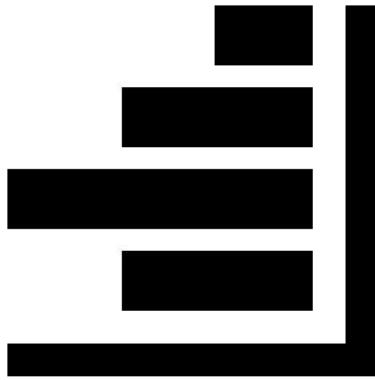


# Risk to Success of the Test – Mitigations Cube

		Mitigation Cube: Success of the Test				
Risk Probability/ Severity	Catastrophic (A)	Critical (B)	Moderate (C)	Minor (D)	Negligible (E)	
5						
4						
3						
2						
1						

- Design Parameters**
- 7-9, 17-18, 23-25, 36-38, 42-43
- Inspection of Test Article/Equipment**
- 11, 16
- Operation**
- 16, 26, 31, 34
- Training**
- 12, 39-42
- Handling of Test Article/Equipment**
- 15
- Storage of Test Article/Equipment**
- 28-30, 44





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# Post-Test Analysis



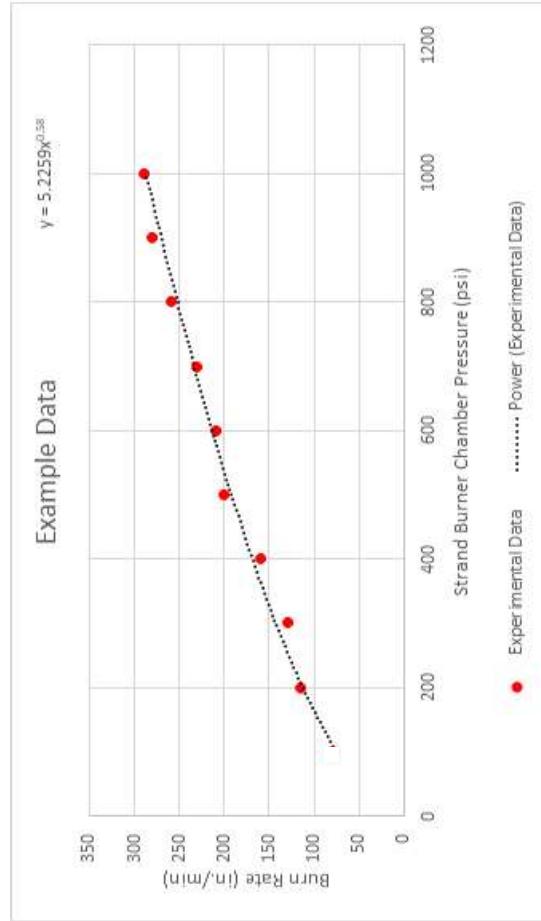
# Post Test Analysis

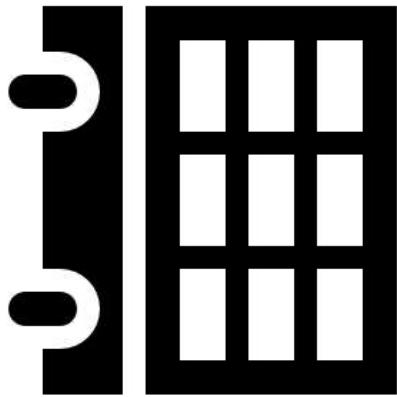
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- Calculate burn rate for each wire section
- Plot the burn rate versus Pressure from all tests
- Obtain ‘a’ and ‘n’ values from power line best fit
- Compare results from MIT data and new collected data
- Perform uncertainty calculations for ‘a’ and ‘n’ values

# Post Test Analysis

- 9 burn rates per test plotted against respective pressures
- Power line of best fit for a and n values
- a and n used to find mass flow rate, thrust, and nozzle design
- Makes testing safer and more efficient





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# Testing Schedule



# Testing Schedule

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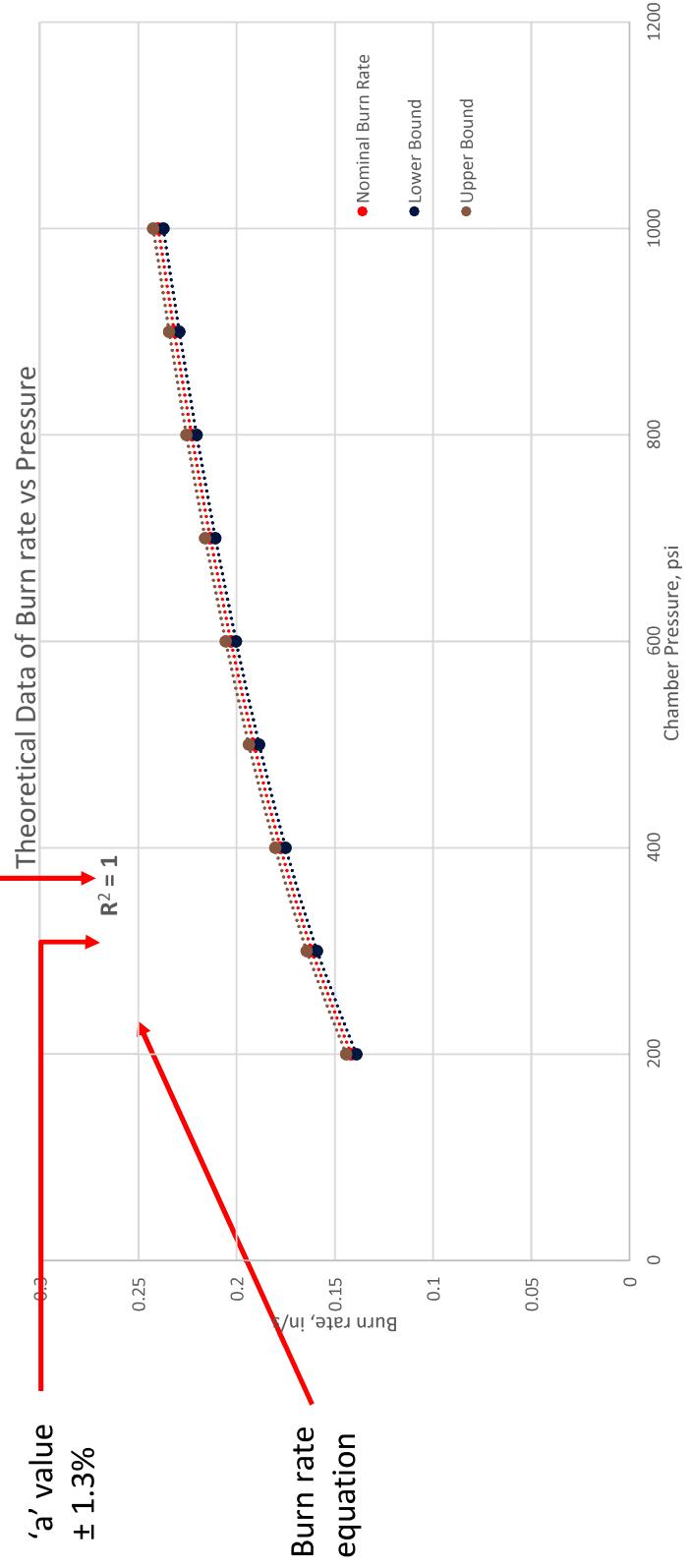
Tests 1-3  
Thursday

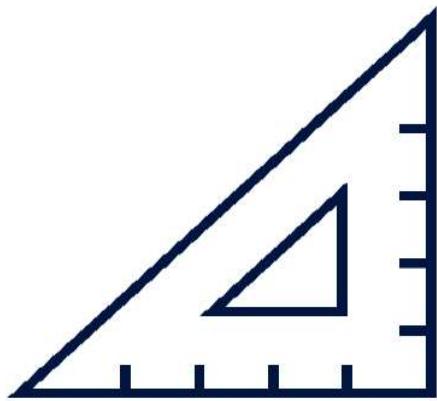
Tests 4-6  
Friday

Tests 7-9  
Saturday



# Instrumentation Analysis





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# Structures



# Subsystem Requirements – Structures

---

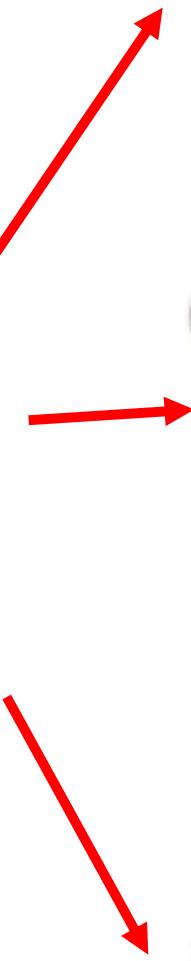
- The strand burner must have a factor safety of 3 [1.1.13]
- Compatible with all current motor sizes build on campus [1.1.2]
- Structurally sound for up to 10 years [1.1.7]



# Subsystem Materials – Structures

---

## Materials



1018 Hot Roll Steel Threaded cap [2]

ASTM A500 Grade B Threaded  
Steel pipe [3]

12L14 Cold Rolled Steel face plate [4]



# Subsystem Governing Equation - Structures

---

Temperature of combustion  
gases in Kelvin [5]

$$T_f = \frac{\Delta u m_p}{C_v} + T_i$$

Final chamber pressure after  
propellant is burnt [5]

$$P = \frac{m_g R_g T_g}{V_{ch}}$$

$T_g$  = Temperature of gases including Argon

$\Delta u$  = Amount of energy released (NASA CEA)

$m_p$  = Mass of propellant

$C_v$  = Specific heat capacity of gases

$T_i$  = Initial temperature = 293 K

$m_g$  = Mass of all gases including argon

$R_g$  = Gas constant (including argon)

$T_g$  = Final temperature in chamber

$V_{ch}$  = Volume of the chamber



# Subsystem Governing Equation - Structures

---

Percent increase in chamber pressure

$$P_{inc} = \frac{p_i - p_f}{p_i} \times 100$$

Temperature of outer surface of  
the pipe [6]

$$T_2 = \frac{\Delta U}{mC_p} + T_1$$

**Maximum Pressure increase is 5%**

$p_{inc}$  = Percent increase in pressure

$p_i$  = Initial chamber pressure

$p_f$  = Final chamber pressure

m = Total mass of structural components

$C_p$  = Specific heat capacity of steel

$T_1$  = Initial temperature of chamber

$T_2$  = Final temperature in chamber

$\Delta U$  = Total energy released



# Pipe Wall Numerical Heat Transfer Analysis

## Assumptions:

- Plane wall transient heat transfer
- Adiabatic (heat does not enter or leave the system) back face with infinite convection coefficient
- Gas temperature obtained from NASA CEA
- All energy released transferred to the pipe

Temperatures in the table are in Kelvin

303.3	303.6	303.9	304.2	304.5	304.9	305.2
303.4	303.7	304.0	304.3	304.6	304.9	305.2
303.4	303.7	304.0	304.3	304.6	304.9	305.2
303.4	303.7	304.0	304.3	304.6	304.9	305.3
303.5	303.7	304.0	304.3	304.6	304.9	305.3
303.5	303.8	304.1	304.3	304.7	305.0	305.3
303.5	303.8	304.1	304.4	304.7	305.0	305.3
303.6	303.8	304.1	304.4	304.7	305.0	305.3
303.6	303.8	304.1	304.4	304.7	305.0	305.3
303.6	303.9	304.1	304.4	304.7	305.0	305.3

90°F

# Subsystem Governing Equation - Structures

---

Minimum thickness of pipe [7]

$$th = \frac{r_i P}{A_S E - 0.6P}$$

Additional thickness due to  
threads [7]

$$t = \frac{0.8}{n}$$

th = Minimum thickness of pipe

$r_i$  = Inner radius of pipe

P = Chamber pressure

$A_S$  = Maximum allowable stress of pipe material

E = Joint efficiency

t = Additional thickness due to threads on the pipe

n = Number of threads per inch



# Subsystem Governing Equation - Structures

---

Moment at point b  
<sup>[8]</sup>

$$M_{rb} = qa^2 \left( \frac{L14}{C5} \right)$$

Stress at point b <sup>[8]</sup>

$$\sigma_{rb} = \frac{6M_{rb}}{th^2}$$

$M_{rb}$  = Moment at point b

q = Pressure

a = Radius of plate

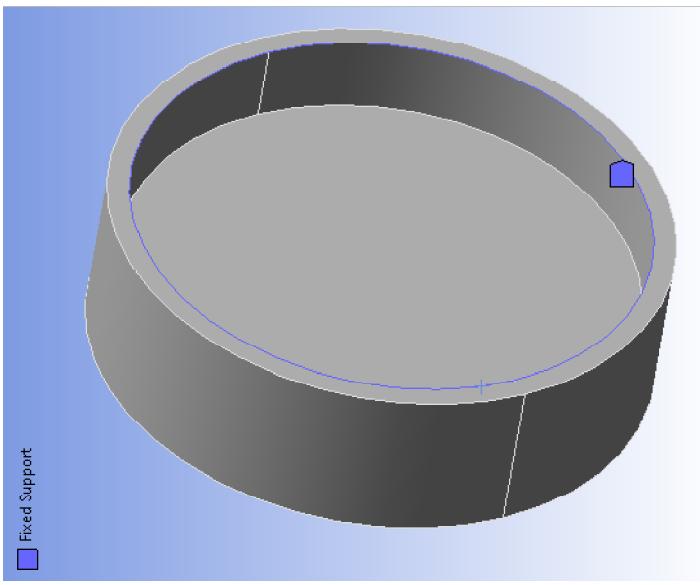
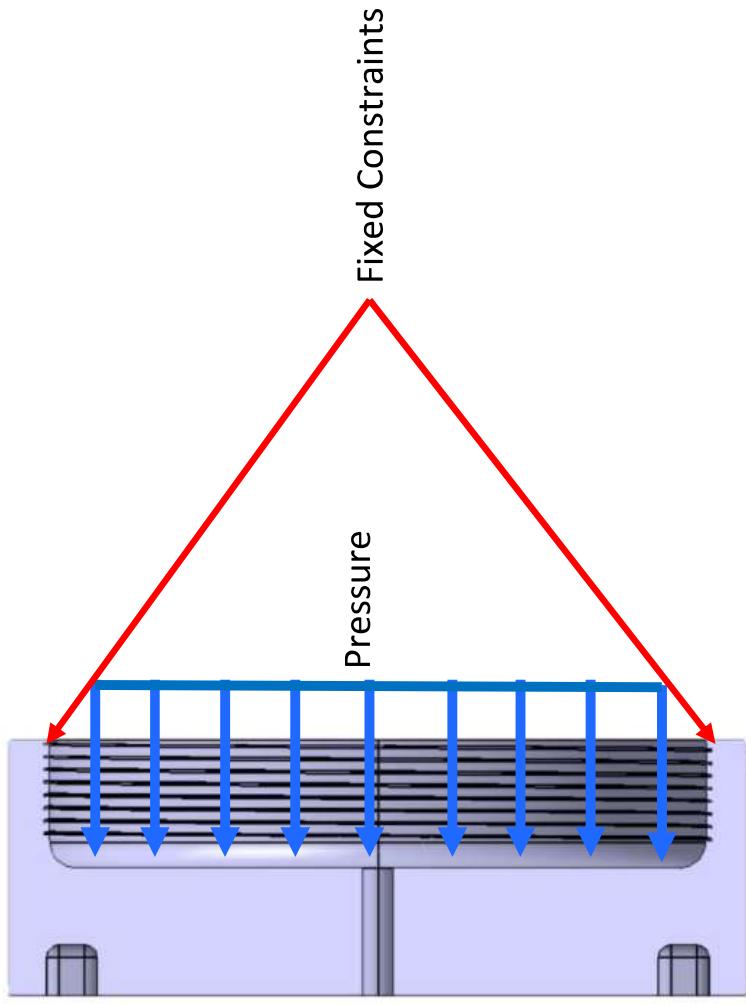
L14 & C5 = Constants

$\sigma_{rb}$  = Stress at point b

th = Minimum thickness of caps

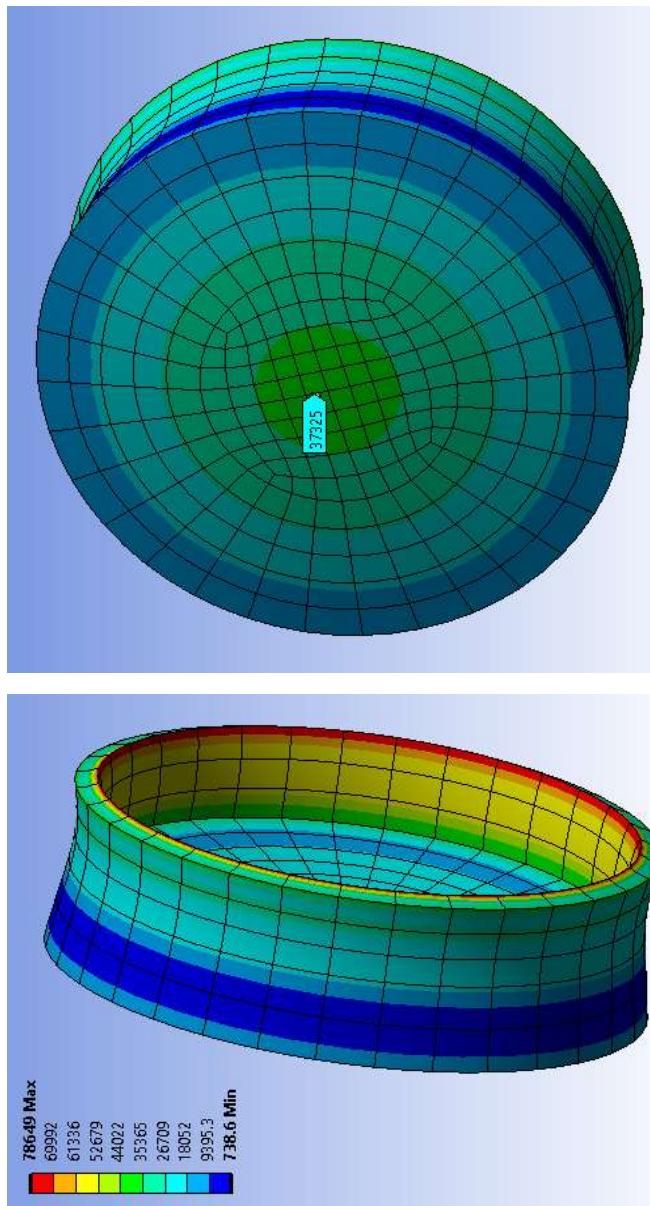


# Numerical Analysis of Threaded Caps



# Numerical Analysis of Cap Thickness

- Maximum stress: 37325 psi
- Hand calcs: 38352 psi
- Tensile strength: 58000 psi



# Subsystem Governing Equation - Structures

---

Number of threads in caps [9]

$$n = \frac{F_p}{0.5\tau_s p \pi d}$$

n = Number of threads

$\tau_s$  = Shear strength

p = Pitch

d = Pitch diameter

$F_p$  = Force on the plate



# Subsystem Governing Equation - Structures

---

Number of cycles before failure [10]

$$a = \frac{(f S_{ut})^2}{S_e}$$

$$b = -\frac{1}{3} \log \left( \frac{f S_{ut}}{S_e} \right)$$

$$N = \left( \frac{\sigma_{rev}}{a} \right)^{\frac{1}{b}}$$



**N = 37430 cycles**

b = Constant

N = Number of cycles before failure

$\sigma_{rev}$  = Completely reversed stress

f = Fatigue strength fraction

$S_{ut}$  = Ultimate tensile strength

$S_e$  = Endurance limit

a = Constant



# Subsystem Governing Equation- Structures

---

Stress in bolts of face plate

$$\sigma = \frac{\pi d^2 p}{4}$$

Tensile strength of 1/4 28  
bolts: **3000psi**

**2863 psi**

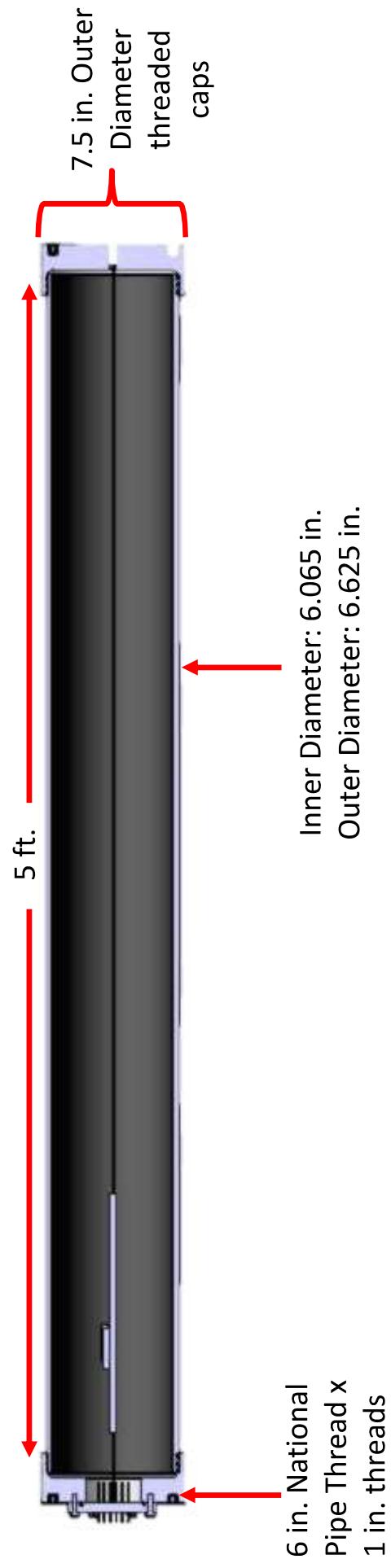


d = Diameter of face plate

P = Chamber pressure

$\sigma$  = Stress in bolts

# Subsystem Dimensions- Structures



# Subsystem Summary- Structures

---

- Strand burner is made of ASTM A500 Grade B pipe with inner diameter of 6 in. and length of 5 ft.

- 1 in. NPT 6 in. threading on both ends of the pipe

- Two 2 in. threaded caps are made of 1018 Hot Roll Steel

Requirements		Number	Title	Value	Analysis Value	Verification Method	Meets Requirements
1.1.13	Factor of Safety for pipe and caps			$\geq 3$	$\geq 3$	Hydrostatic Test	TBD



# Pressure System



# Subsystem Requirements – Pressure System

---

The Pressurization Panel must meet these requirements:

- Pressurization Panel will be proofed to 1.5x MEOP; 4.5 ksi. [1.1.14]
- Must deliver Argon to valve actuators at a pressure of 80 psi, but no more than 120 psi.
- Must deliver Argon to Strand Burner at a pressure of 1000 psi.
- Methods to relieve pressure from system in event of over-pressurization

The Strand Burner must meet these requirements:

- Strand Burner will be proof tested to 3x MEOP; 3 ksi. [1.1.13]
- Must be able to vent pressure from Strand Burner remotely



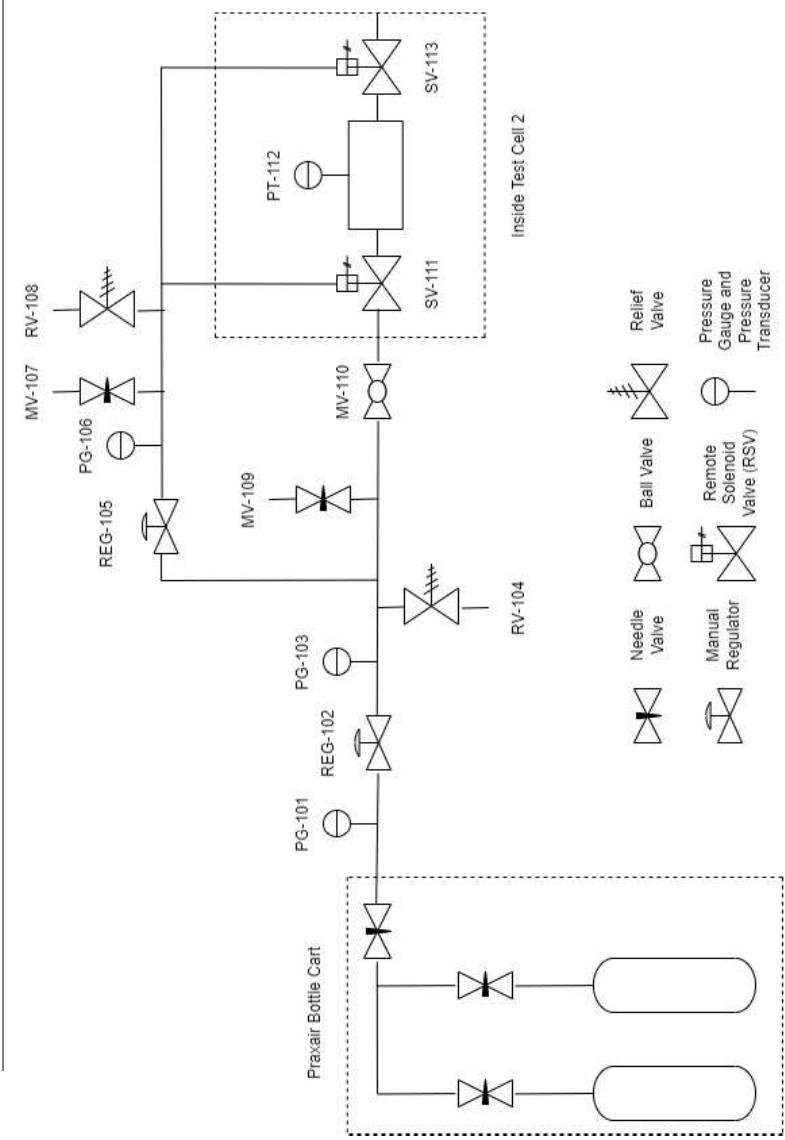
# Subsystem Materials – Pressure System

---

- $\frac{1}{4}'' \times 0.049''$  S.S. Tubing
- High Pressure Regulator with 6 ksi. input/2 ksi. output
- Muscle Pressure Regulator with 6 ksi. input/200 psi. output
- $\frac{1}{4}''$  Needle Valves rated to 6 ksi.
- $\frac{1}{4}''$  Ball Valve rated to 6 ksi.
- High Pressure Relief Valve set at 3 ksi.
- Muscle Pressure Relief Valve set at 125 psi.
- Pressure Transducer rated to 3 ksi.



# Subsystem Integration – Pressure System



Unless otherwise specified,  
components will be located on  
the pressurization panel

The electronically actuated valves  
will be inside Test Cell 2 and  
connect to the PXle located in TC3

Regulators control the pressure  
level to the valves and strand  
burner



# Hazardous Operation Assessment Definition

---

## Severity Ratings:

- 1 – Not a risk to system/personal/data and/or minor time loss
- 2 – Loss of consumables and/or minor to no injury to personnel and/or major time loss
- 3 – Damage to component and/or minor injury to personnel and/or loss of data
- 4 – Damage to system and/or loss of component and/or severe injury to personnel
- 5 - Complete loss of article and/or loss of life

## Likelihood Ratings:

- 1 - Will most likely never occur in the life of the system
- 2 – Likely to occur once every 100+ uses of system
- 3 - Likely to occur once every 50+ uses of system
- 4 - Likely to occur once every 10+ uses of system
- 5 - Guaranteed to occur with every use of system

S	L	1	2	3	4	5
1	1	2	3	4	5	5
2	2	4	6	8	10	
3	3	6	9	12	15	
4	4	8	12	16	20	
5	5	10	15	20	25	



# Risks – Pressure System

---

- Manual Iso (MV-110): Misconfigured Closed
- Remote Press (SV-111): Loss of control
- Remote Vent (SV-113): Loss of control
- **Manual Vents (MV-107,109): Misconfigured Open**
- REG's (REG-102,105): Fail full open

Likelihood \ Severity	1	2	3	4	5
1	Green	Green	Green	Yellow	Red
2	Green	Green	Yellow	Yellow	Red
3	Green	Yellow	Yellow	Red	Red
4	Green	Yellow	Red	Red	Red
5	Yellow	Red	Red	Red	Red

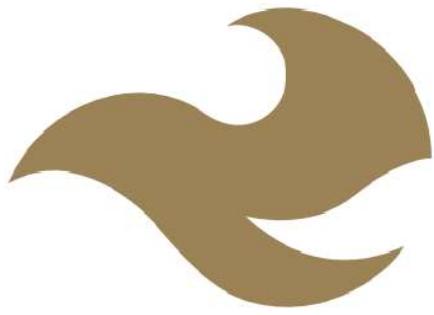


# Mitigation – Pressure System

- Manual Iso (MV-110): Procedures, Training
- Remote Press (SV-111): Fail state closed, procedures, isolation
- **Remote Vent** (SV-113): Proofing, fail open
- **Manual Vents** (MV-107,109): Procedures, training
- **REG's** (REG-102,105): Proofing, isolation, pressure visibility, relief valves.

Likelihood \\ Severity	1	2	3	4	5
1	Remote Press				
2	Manual Iso, Remote Vent, Manual Vents, REG's				
3					
4					
5					





# Ignition



# Subsystem Requirements – Ignition

---

The ignition system must satisfy requirement [1.1.5]:

- Must ignite propellant 90% of the time



# Subsystem Materials – Ignition

---

- Electric Matches
- Ignition Switch Box
- Continuity Lights
- Ignition and Continuity Wiring



# Subsystem Analysis – Ignition

---

## Benefits:

- Highly versatile
- Easy to implement and use
- Good compromise between cost and reliability
- Existing protocols that address safety concerns

## Concerns:

- Premature ignition
  - Mitigation: Circuit continuity lights & control lockout key



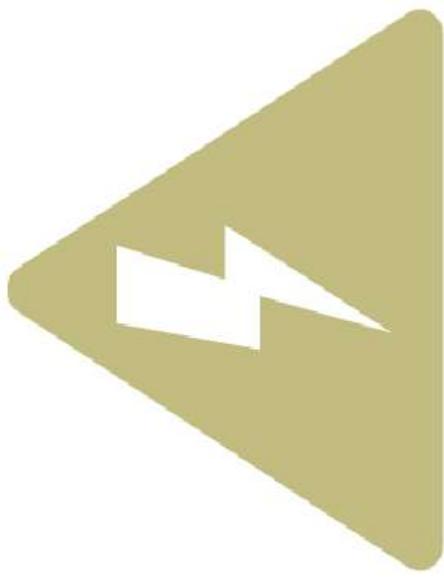
# Manufacturing Plan – Ignition

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The manufacturing plan for the ignition system consists of two parts:

- Propellant Integration
  - Match is placed in propellant jig during the propellant casting processes
- Electrical System
  - Electric connector/wires for match are routed out of strand burner via pass-throughs
  - Wiring will connect to ignition switch box
  - Continuity lights will be installed in the switch box and/or wiring that show when the circuit is hot





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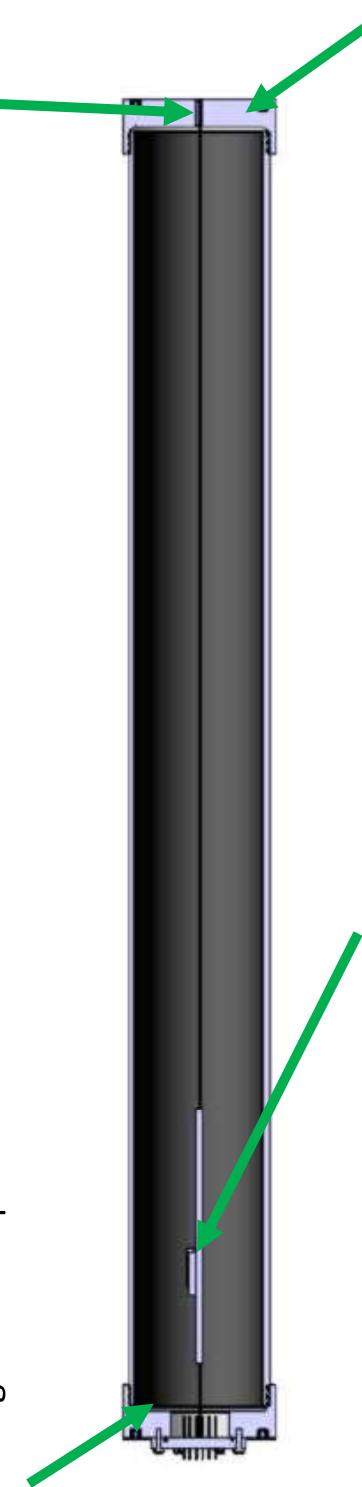
# Burn-Away Solder Wires



# Subsystem Overview – Burn-Away Solder Wires

The wires are passed through the passthrough in the cap

Pressure system connection



The propellant with burn away wires sits on an internal plate

Pressure Transducer screwed into cap

# Subsystem Requirements – Burn-Away Solder Wires

---

The subsystem must meet requirement [1.1.1] :

- Must be able to support a 100 Hz. measurement speed
- Must have a 90% confidence interval for data collected



# Materials – Burn-Away Solder Wires

---

Material	How to acquire?	Dimensions
Fire Lizard Propellant	Produced on Campus	0.25 in. X 0.25 in. X 2 in.
Propellant Jig	Machined	0.25 in. X 0.25 in. X 2 in.
Plexiglass	Purchase	N/A
Thin Solder Wire	Purchase	0.3 mm.
Sealant	Purchase	N/A
Mold Release	Purchase	N/A

# Subsystem Governing Equation – Burn-Away Solder Wires

---

Discrete linear burn rate [12]

$$\dot{r}_i = \frac{l}{t}$$

Average burn rate [12]

$$\dot{r}_{avg} = \frac{\sum \dot{r}_i}{\# r_i' s}$$

$\dot{r}$  = Burn Rate

$l$  = Distance between wires

$t$  = Time between wire breaks



# Subsystem Analysis - Burn-Away Solder Wires

---

- The burn rate is calculated for each of the individual lengths and averaged
- The burn rate is plotted against the corresponding pressure
- A series of 9 tests is conducted before one can calculate the  $a$  and  $n$  values

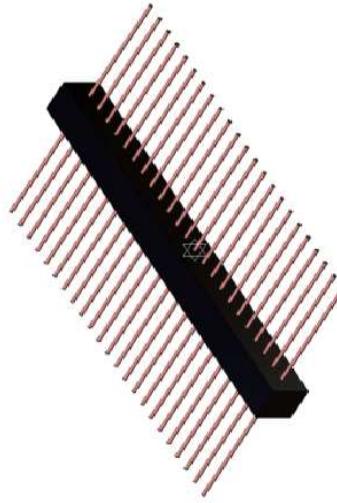


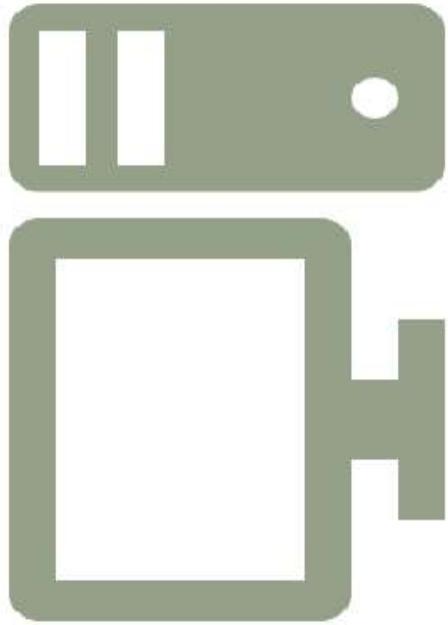
# Manufacturing Plan- Burn-Away Solder Wires

---

The manufacturing plan for this system includes two parts:

- The propellant itself:
  - Made in small batches using the RDL mixing procedures
  - Fire Lizard Recipe (Eagle Space Flight team working propellant)
  - Poured into jigs for consistency
- The propellant jigs consists of:
  - Plexiglass
  - Sealed with epoxy
  - Coated with mold release
  - Wire Slots





---

# Data Acquisition Software and Interfacing



# Subsystem Requirements – Data Acquisition Software and Interfacing

---

Subsystem must satisfy the following requirements:

- The data acquisition system must interface with the *PX/e* and Test Cell 3
- Continuity light must turn on while connected to the ignition circuit [1.1.8]
- Lockout key must be used in the ignition box to prevent premature ignition

[1.1.4]

[1.1.9]



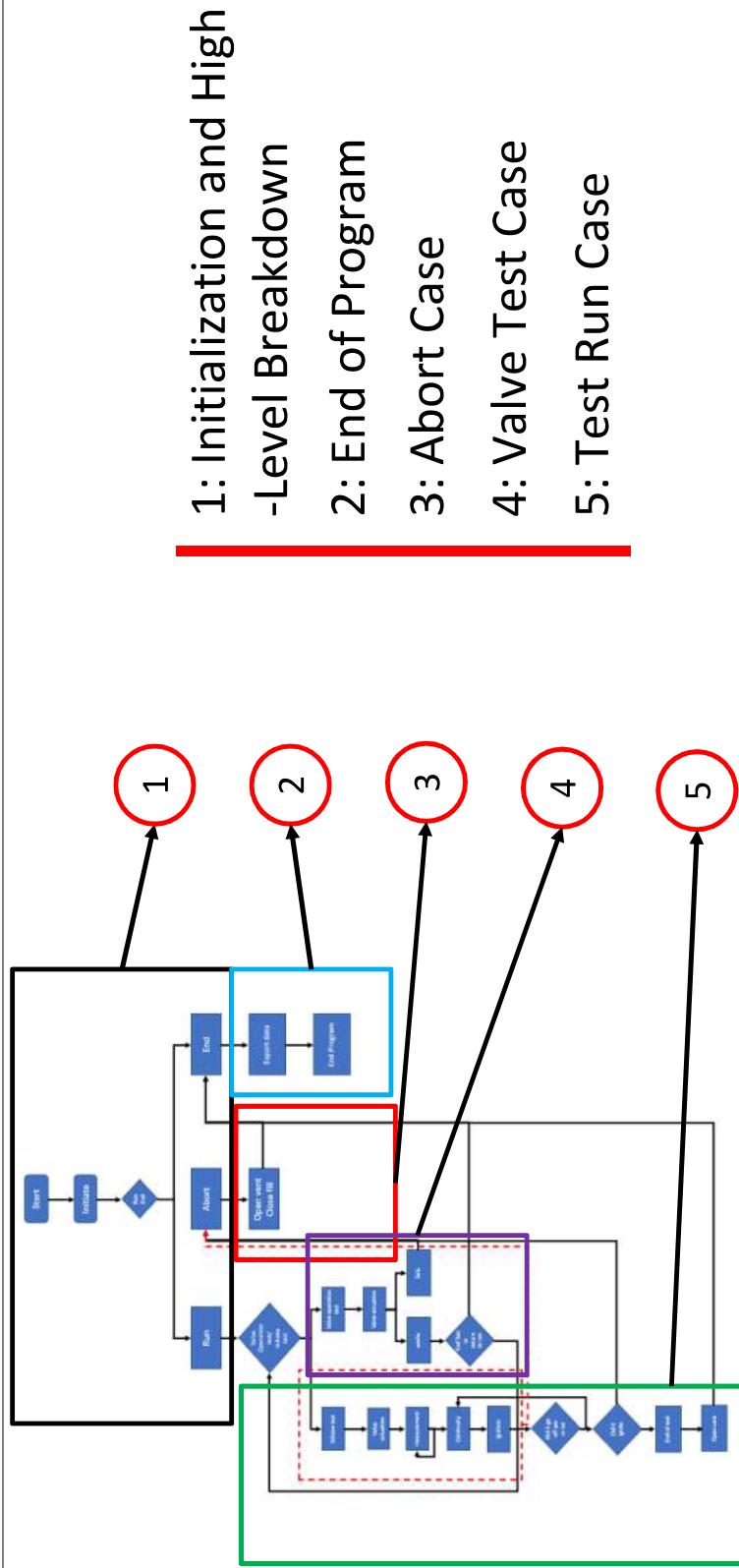
# Materials– Data Acquisition Software and Interfacing

---

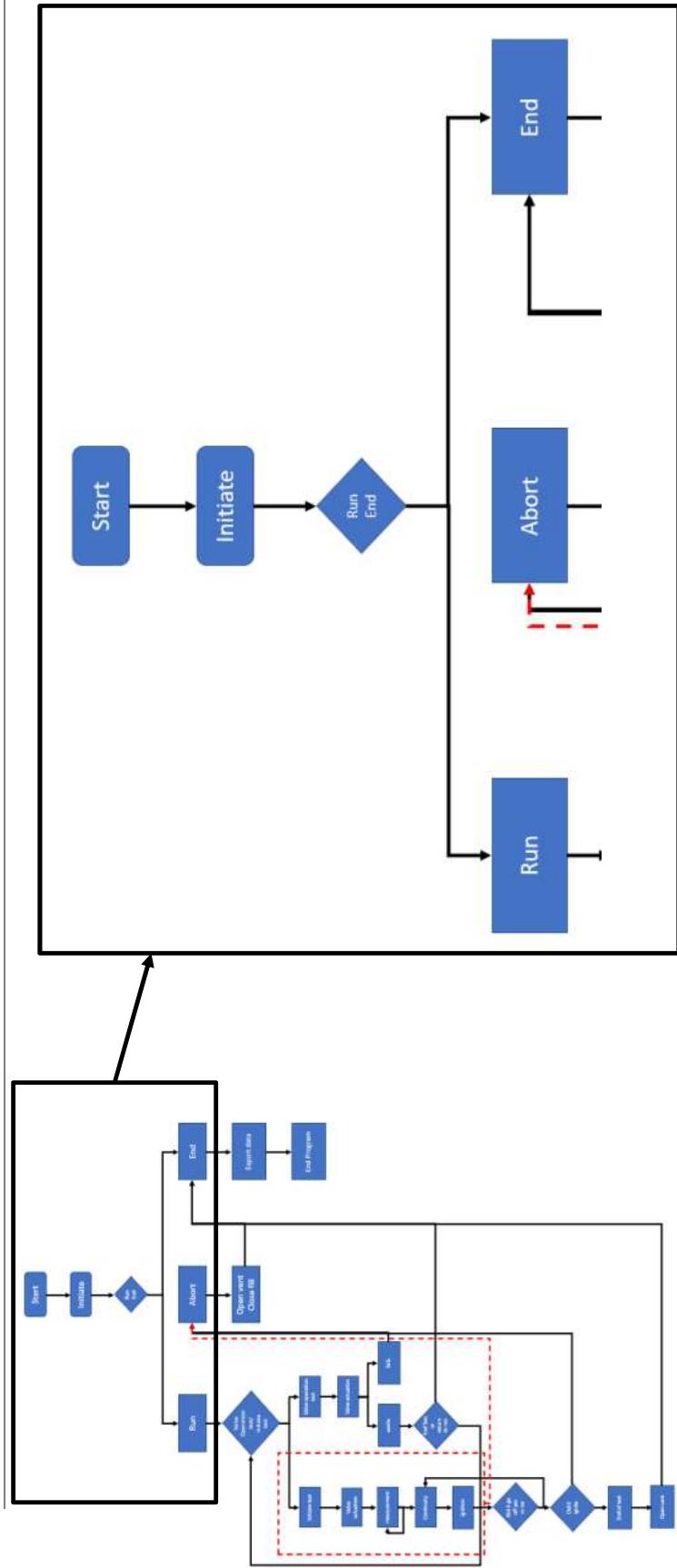
Material	How to acquire?	Dimensions
Connection Wires	Produced on Campus	N/A
PXIe	Customer Supplied Equipment	N/A
LabVIEW	Customer Provided Code	N/A
NEMA 4 Rated Box	Purchase	20x16x6
Catwalk	Purchase	N/A
Cable Connections	Purchase	N/A



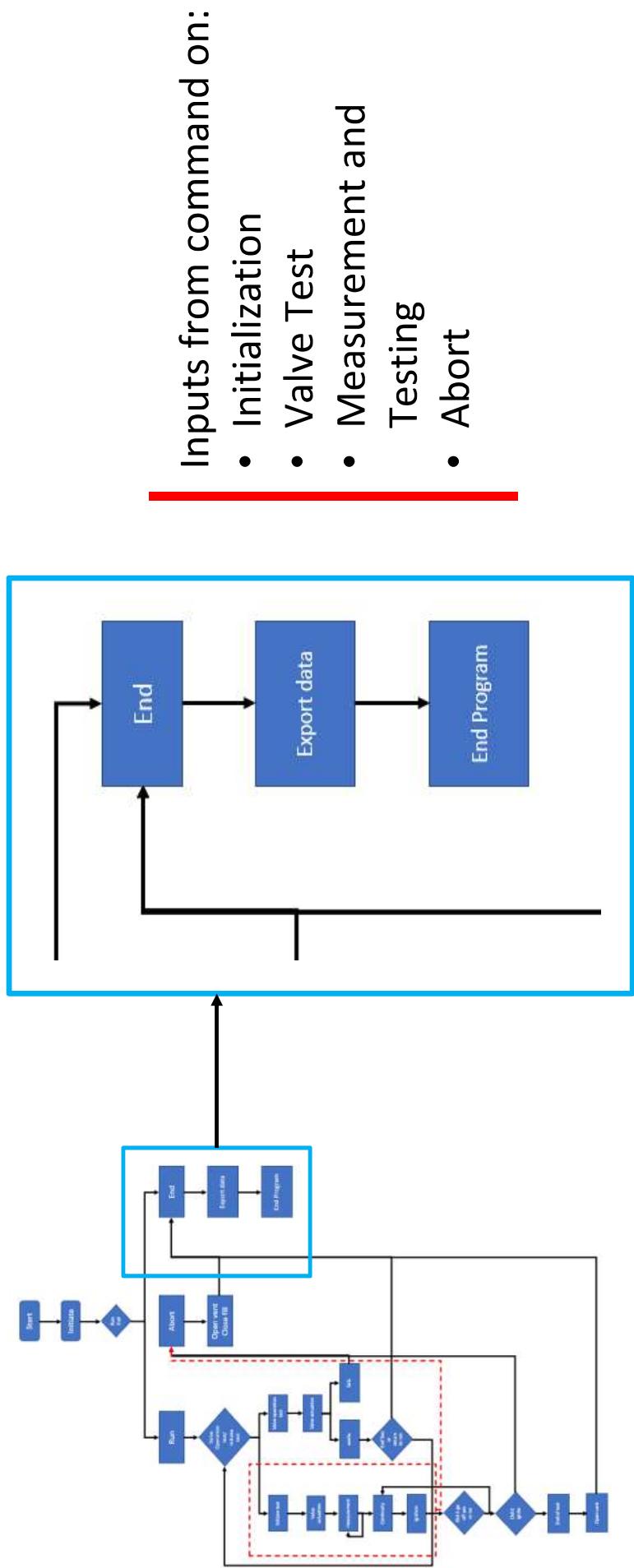
# Subsystem Integration – Control Code Flow Diagram



# Control Code Flow Diagram- 1: Initialization and High-level Breakdown

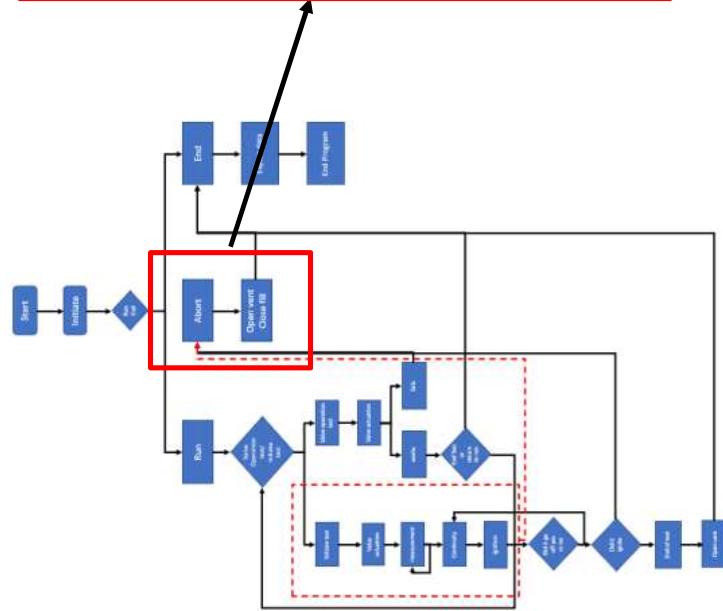
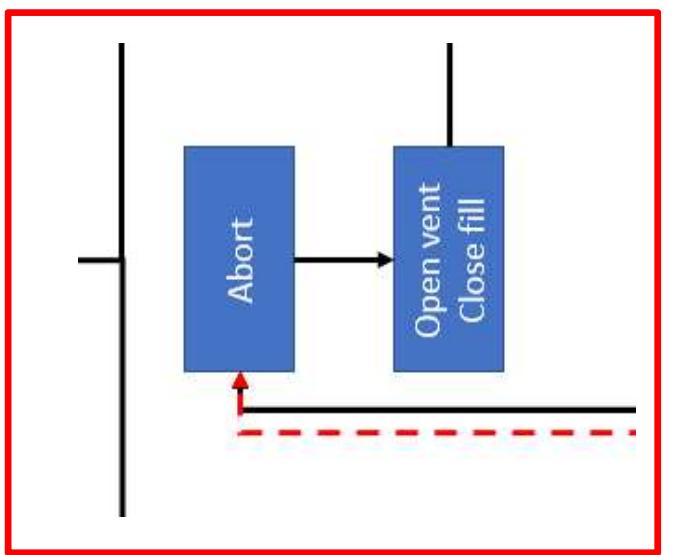


# Control Code Flow Diagram- 2: End Case



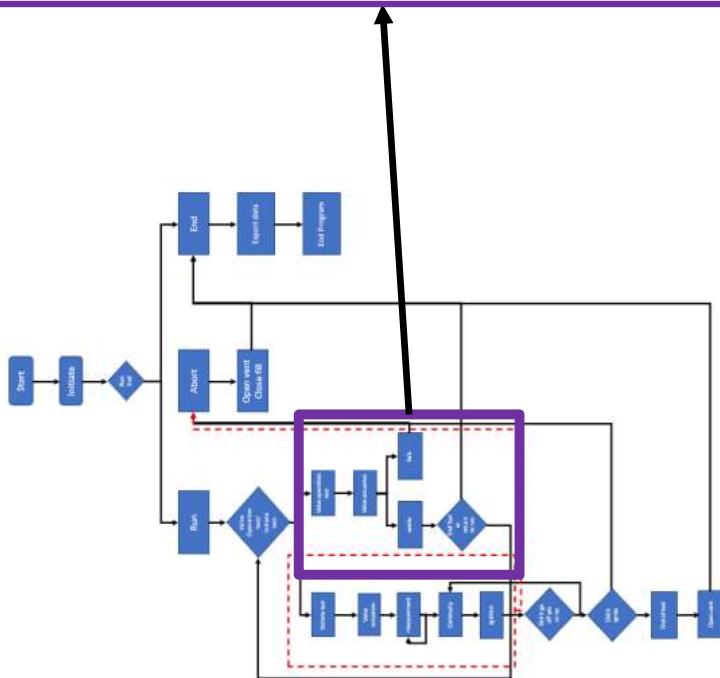
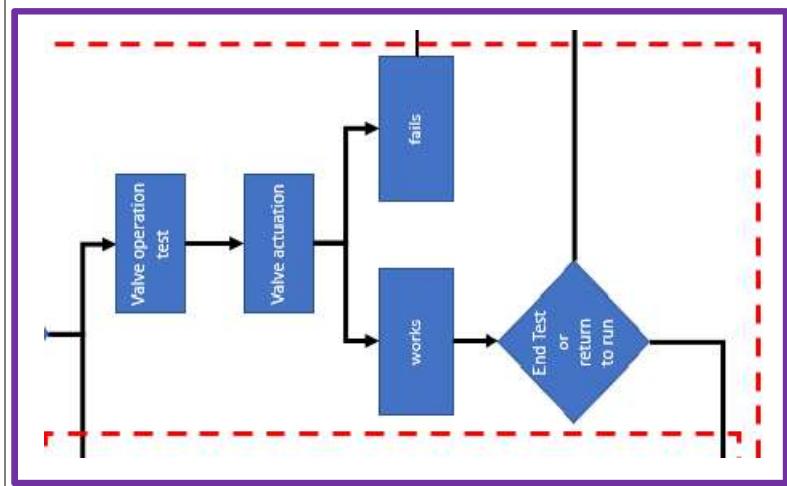
# Control Code Flow Diagram- 3: Abort Case

- Inputs from command on:
- Manual Button
  - Valve Test
- Output:
- End
  - Reset

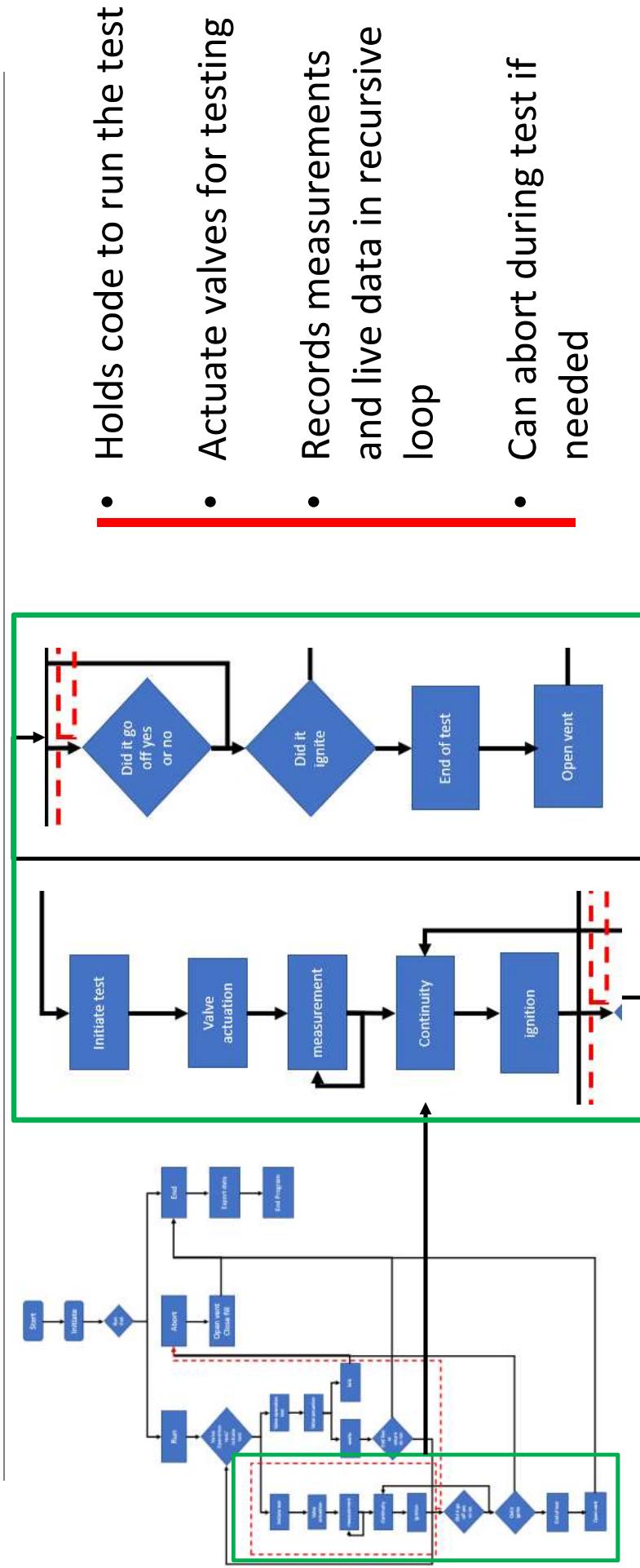


# Control Code Flow Diagram- 4: Valve Test Case

- One branch of the Run Case
- Allows for pre-test verification of valve operation
- Allows for user to end program or to continue to testing



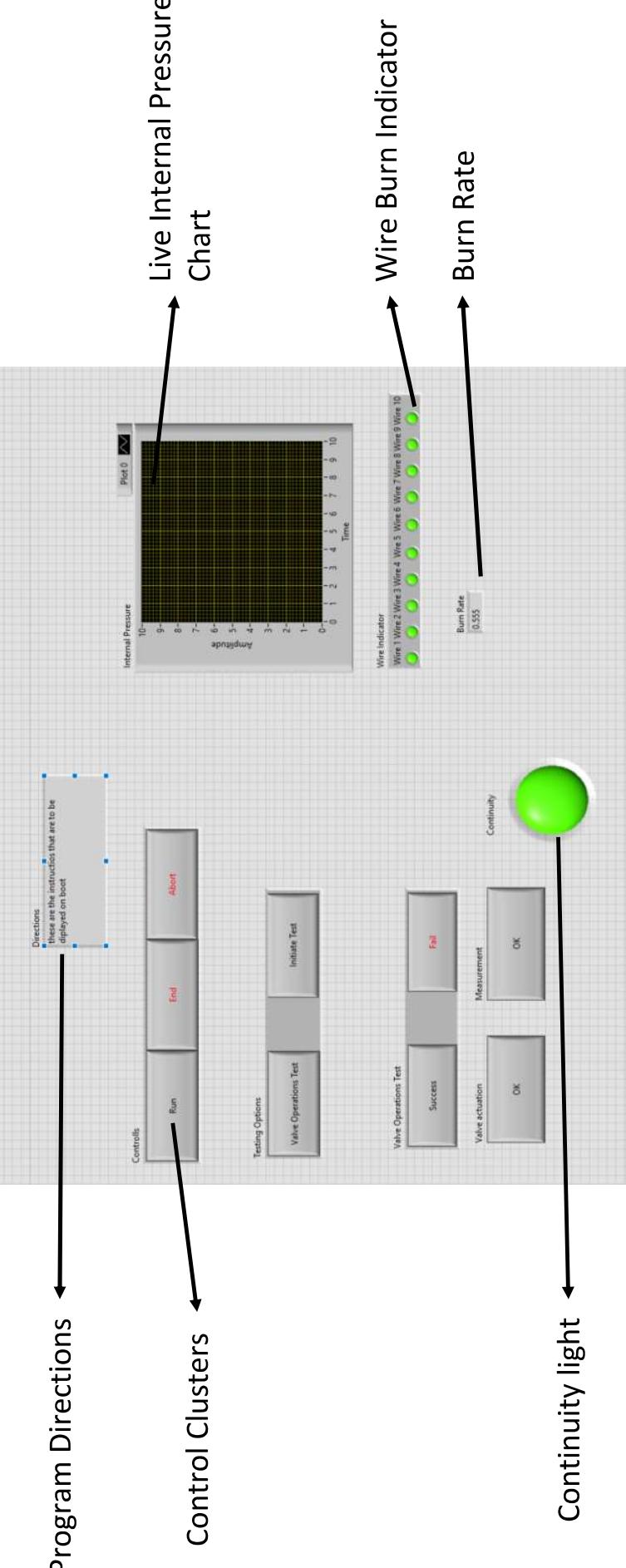
# Control Code Flow Diagram- 5: Test Run Case



)



# Subsystem Integration – Control Code Interface Panel



# Drawing Trees

Complete and Approved

In progress

Not yet started

ME431-S22-01-SKD-  
00-100  
Strand Burner Assembly

ME431-S22-01-SKD-00-101  
Pipe

ME431-S22-01-SKD-00-102  
Cap 1

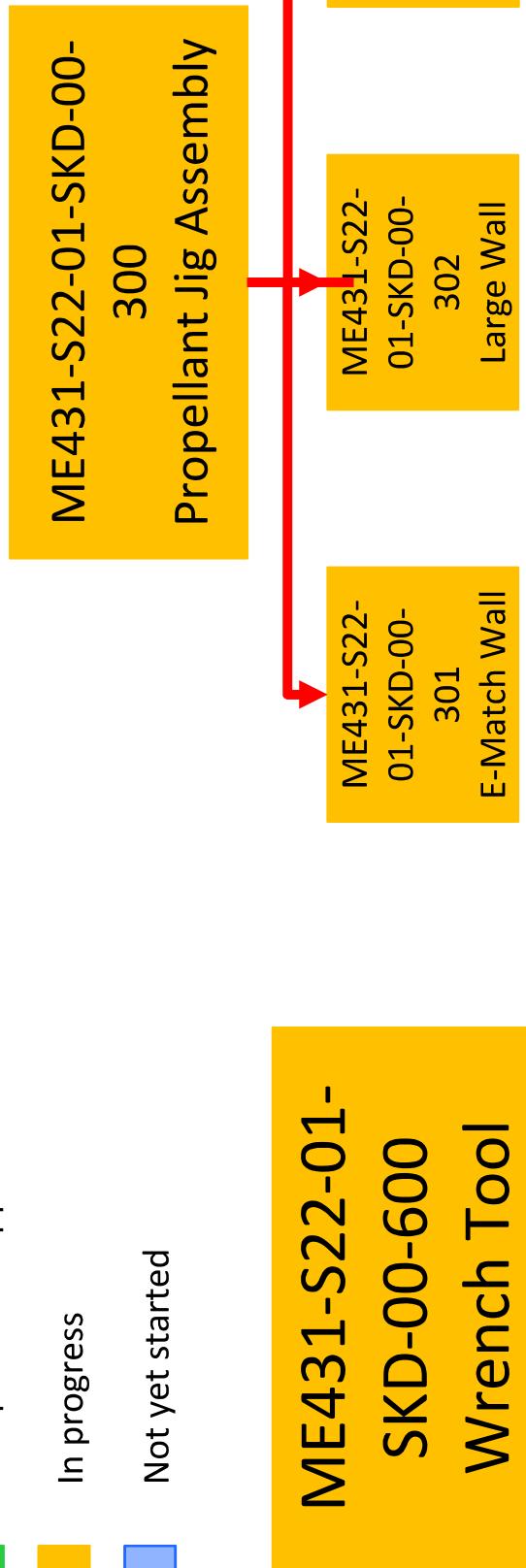
ME431-S22-01-SKD-00-103  
Cap 2

ME431-S22-01-SKD-00-105  
Feet



# Drawing Trees

- Complete and Approved
- In progress
- Not yet started



# Drawing Trees

Complete and Approved

In progress

Not yet started



ME431-S22-01-SKD-  
00-500  
Pinout Box

ME431-S22-01-SKD-00-  
200  
Valve Assembly

ME431-S22-01-SKD-  
00-400  
Pressure Panel

ME431-S22-01-  
SKD-00-201  
Valve Stem

ME431-S22-01-  
SKD-00-202  
Valve Standoff



# Manufacturing Plan

---

## Step 1

Place an  
order of  
materials

## Step 2

Receipt of  
ordered  
materials

## Step 3

Outsource  
pipe for  
threads

CNC caps  
out of  
round bar

Cut out  
propellant  
jig

Water jet  
pressure  
panel



# Manufacturing Plan

---

**Step 4**

Cut out Valve  
stems

**Step 4**

CNC Face  
plates &  
thread caps

**Step 4**

Mill wrench  
tool & Pinout  
Box



# Manufacturing Plan

---

**Step 5**

Standoffs for  
valves & tap  
caps

**Step 5**

Machine Feet  
& Shrink fit  
wrench tool

**Step 5**

Install wires  
and seal them  
in Epoxy



# Manufacturing Plan

---

**Step 6**

Coat the Pipe  
interior

**Step 6**

Coat the Caps  
interior

**Step 6**

Install the  
NEMA box in  
TC3



# Bill of Materials

# Bill of Materials

Pin-Out Board Part Numbers

# Testing Overview

Testing Plan



Con-Ops



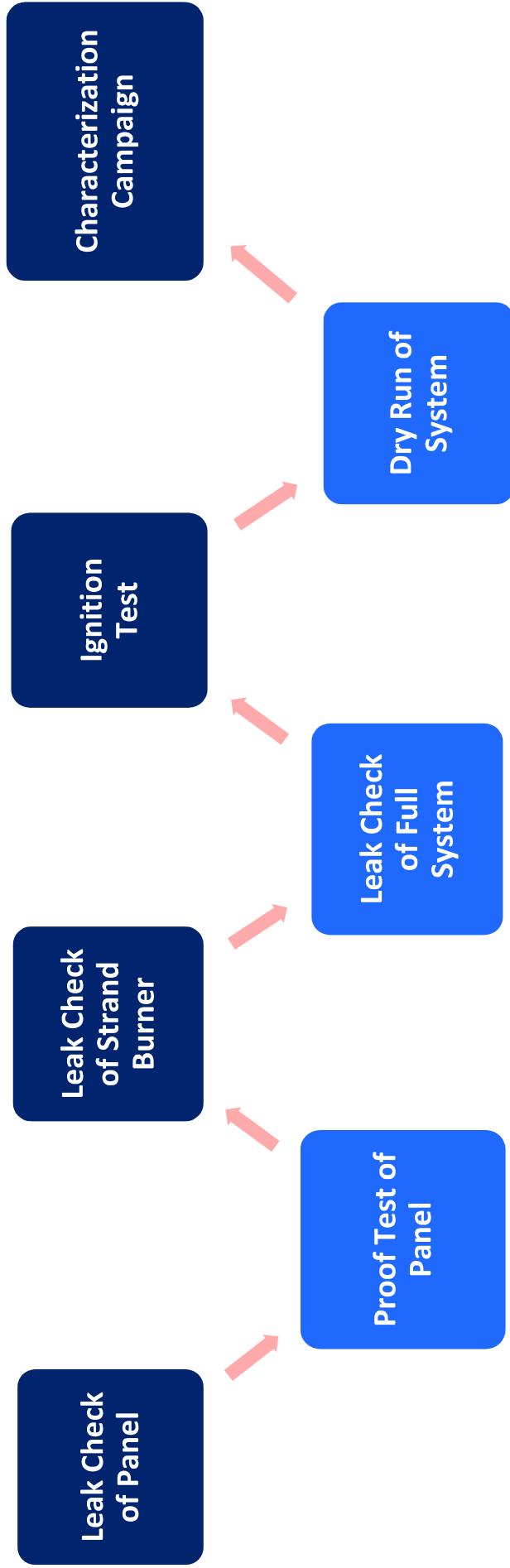
Facilities and Equipment



Instrumentation Plan



# Testing Plan



# Overview/Con-Ops of Verification Tests

---

## Leak Check:

Verification of seals and torquing with 100psi. Argon

1. Close-up system
2. Attach argon bottles
3. Open argon bottle halfway
4. Set reg to ~100 psi.
5. Allow system to pressurize and look for leaks
6. Once all leaks are sealed, close argon bottles
7. Vent system
8. Close-out system



# Overview/Con-Ops of Verification Tests

---

## Proof Testing:

Verification of pressure 1.5x or 3x above rating with water

1. Close-up system
2. Attach hydrostatic pump
3. Fill system with water with highest vent open to release all air
4. Turn on pump to 500 psi.
5. Increment pressure to 4500 psi. or 3000 psi. Hold for 10 min.
6. Turn off pump
7. Drain system
8. Close-out system



# Overview/Con-Ops of Verification Tests

---

## Ignition Test:

Verify E-match ignites propellant, ig sense and ignition circuit work correctly

1. Connect igniter to ignition circuit
2. Insert igniter into propellant
3. Spark igniter
4. Perform 3 times, if all successful then testing is complete



# Overview/Con-Ops of Verification Tests

---

## Dry Run:

Go through procedures as if doing actual test

1. Perform leak check
2. Perform click checks
3. Insert test specimen into the test article, leave igniter outside
4. Close-up test article
5. Pressurize system
6. Spark igniter
7. Vent System
8. Close-out system



# Test Facilities

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## Test Cell 2:

- Houses Strand Burner

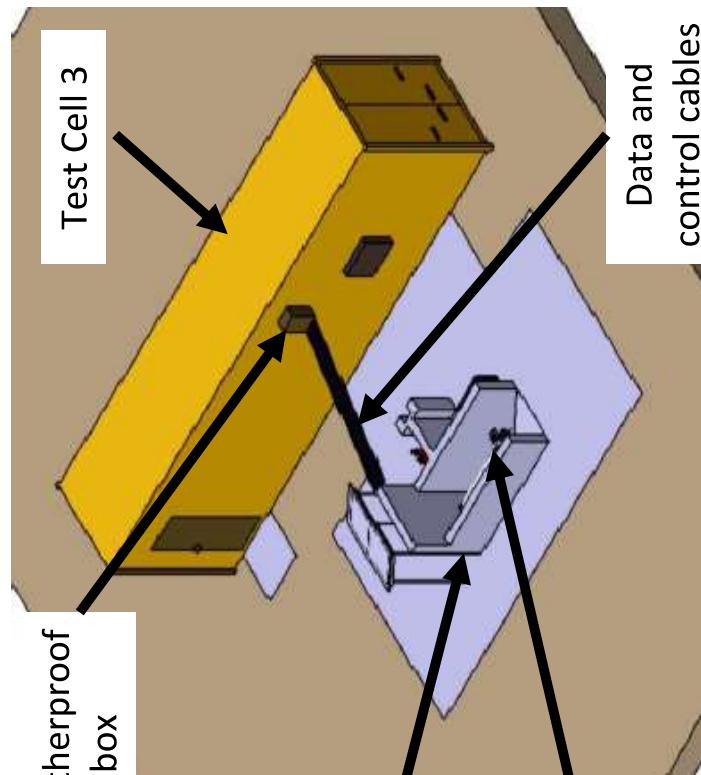
## Test Cell 3:

- Houses PXle
- Stores DAQ Cables

## Building 91:

- Main Control Room
- Operate and Control Strand Burner

# Facilities and Equipment



Hydrostatic Test Pump



[14]

# Instrumentation Plan

What we want?	How to acquire?	Signal Type	Location	Storage: Physical	Storage: Digital
Burn Rate	Talk Back	Digital	Prop Strand		
PX309-3KGV Pressure Transducer		Analog	On strand burner		Data Write to Excel, RDL One Drive, Physical Hard Drive
Pressure Gauge 4089K81 (0-200 psi) 4003K61 (0-1500 psi) 4089K81 (0-4000 psi)		Analog	In line around regs	Trailer /TC3	
Valve State	DB22C-A1RC Limit Switch	Digital	Attached to valve		
Continuity	Ig-Sense	Digital	Igniter, PX/e		



# Conclusion and Compliance Summary

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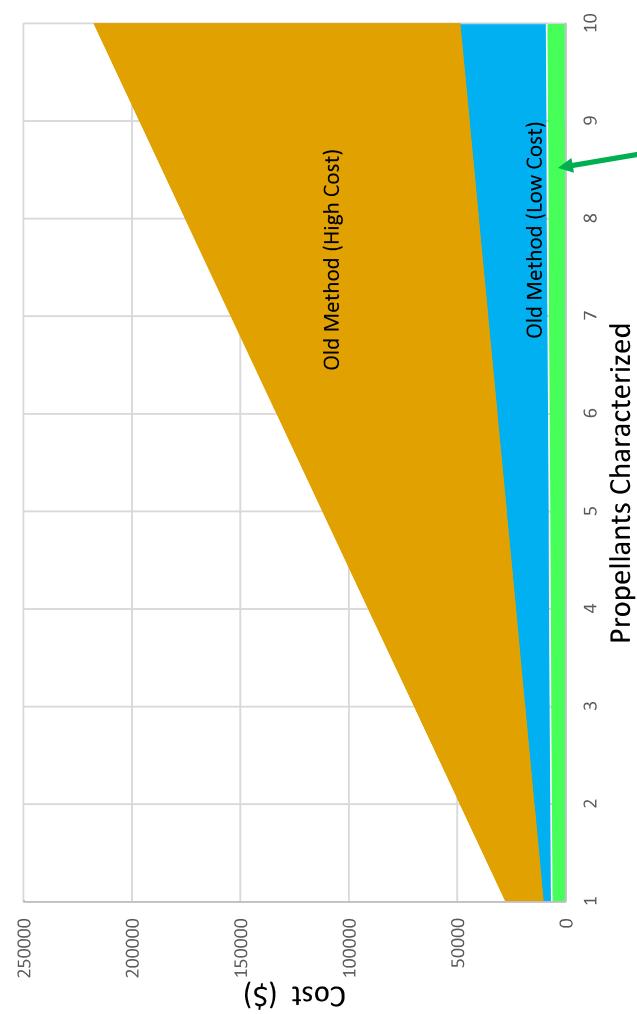
This project delivers:

- Permanent test article to efficiently characterize solid rocket propellants at ERAU
- Reduces the costs by more than half
- More accurate data collection
- Meets the safety requirements of ERAU for similar projects



# Cost Comparison

Cost Comparison

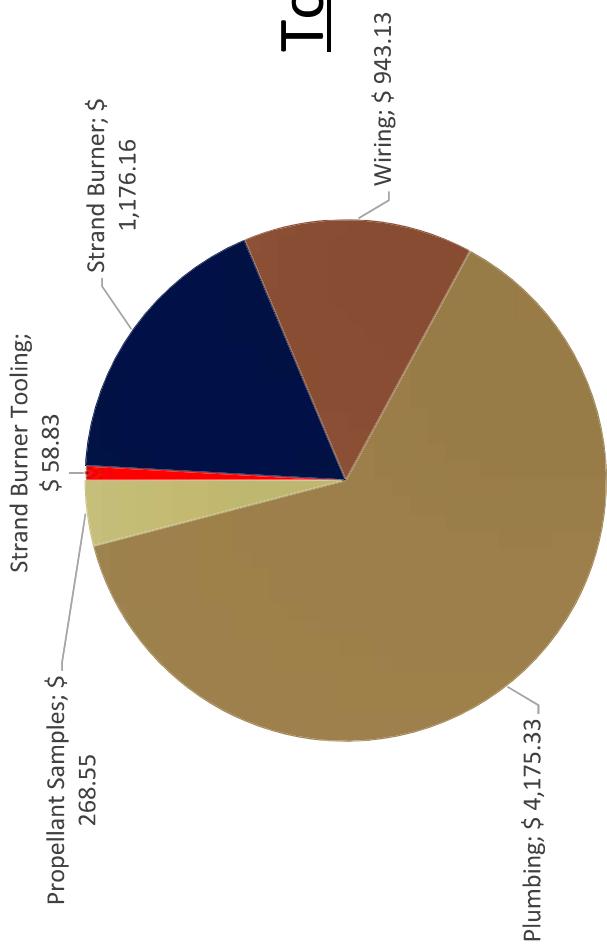


Method	Cost per characterization	Cost at year 1	Cost after 9 propellants characterized
Old Method (High Cost)	\$16,834.88	\$16,834.88	\$168,348.80
Old Method (Low Cost)	\$4,016.94	\$4016.94	\$40,169.40
New Method	\$259.30	\$6459.37	\$8793.03



# Expected Cost Section Breakdown

Expected Cost By Section



■ Strand Burner Tooling   ■ Strand Burner   ■ Wiring   ■ Plumbing   ■ Propellant Samples



# Source of Funding

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Requested Budget	Funding Source
\$2500	College of Engineering Excellence Fund
\$1175	Capstone General Budget
\$2500	Mechanical Engineering Fund for Excellence
\$600	Rocket Development Fund (RDF)
Total: \$6,775	



# Major Risks to the Project

---

- Operations Risks
  - Fabrication/Parts availability
  - Disapproval of grant applications for required funding
  
- Scheduling Risks
  - Long lead times for acquiring parts/supplies
  - Safety approval of test operations



# Major Risks

---

Risk Number	List of Potential Hazards Present Within the Defined Facility Limits
4	Trip/fall over data, power, or other cords or wires
10	Accidental pressing of the ignition button (with the igniter inserted)
13	Accidental pressing of the ignition button (igniter not inserted)
14	Improper instillation of the snap ring causes it to be liberated
15	Excessive wear and tear on the case causing shrapnel or the snap ring to be liberated
20	Dropping of the case
23	Over pressurization causes loss of test article due to rapid unscheduled disassembly of test article



# Major Risks

Risk Number	List of Potential Hazards Present Within the Defined Facility Limits
25	Impact hazard to operational spectators
29	Ignition to surroundings due to flying shrapnel caused by rapid unscheduled disassembly of the motor
38	Over pressurization due to failure of regulator valve
43	Damage to electrical connections due to weather
45	Damage to strand burner due to rust
46	Failure of igniting propellant
54	High lead time on receiving materials and/or machining processes



# Major Risks

L	S	1	2	3	4	5
1			15,16 24,26	25,7,17, 24,26	21,42,50, 1	
2			54,6,18, 33,34,35, 36,48,49		6,38,11,12, 30,31,32,3 8,39,40	
3		13,5,9,13,4 4,53	43,47	20,19,27	10,29,23, 22,28,37, 41	
4			4,3	14,1		
5	46	45,2,8,52				



# Mitigations

Risk Number	Corrective / Preventive Action Recommended
4	Coiling the cords and securing them away from the walkways
10	Having a safety key for the ignition system, design a new ignition box with built in safety interlocks
13	Having a safety key for the ignition system, design a new ignition box with built-in safety interlocks
14	Safety glasses and verbal communication about installation of snap ring
15	Visual inspection of ring and case for defects
20	Properly secure to a rolling cart with ratchet straps during transportation
23	Ensure the case can handle higher than design point pressures and throat diameter is larger than minimum calculated diameter



# Mitigations

---

Risk Number	Corrective / Preventive Action Recommended
25	Trail spotter makes sure all operational personnel are in the designated observation point
29	Have a margin of safety higher than 0.5, and have the test article inside of the test cell, and have the case be made from aluminum
38	The strand burner shall have an electronically actuated ball valve to release the pressure at 2500 psi
43	The data interface box on test cell 3 will be NEMA 4 rated to protect the connections from the weather
45	The strand burner will be removable and able to be stored in TC3
46	Strand burner must be vented and reset. The ignition system must be tested in the empty, pressurized strand burner
54	Utilize multiple sources

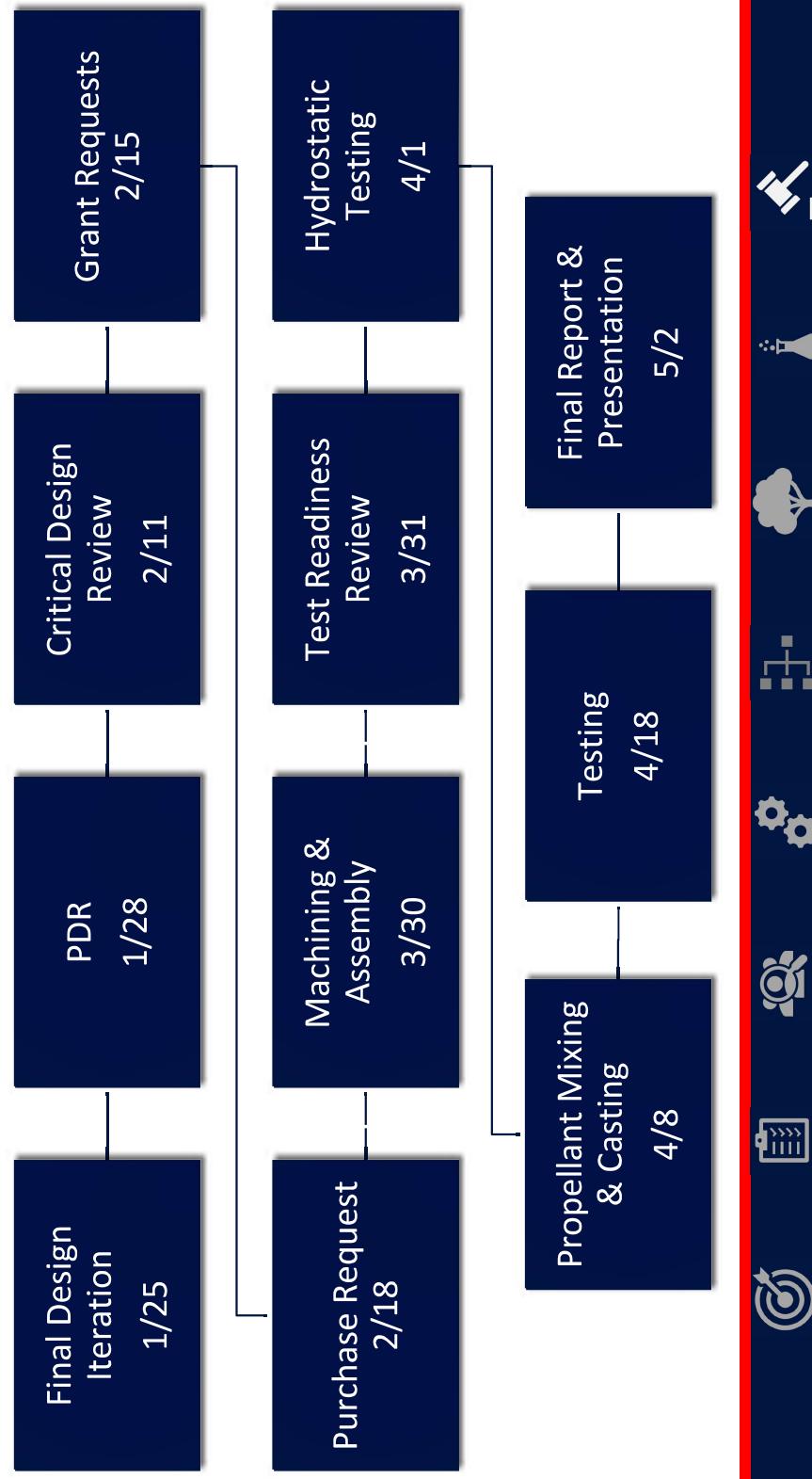


# Mitigations

S L	1	2	3	4	5	
1	13,2,3,5,4 4,53	6,15,43,15,16, 18,33,34,35,3 6,43,48,49	25,7,17, 24,26, 27	10,23,29,38,11, 12,21,22,23, 28,29,30,31,32, 37,39,40,41, 42,50,51		
2		4,45,9,52	20,54,47			
3	46,19	14,1,8				
4						
5						



# Project Timeline



# Back-up Slides

# System Overview

use this so that the bullets point at there corresponding parts

Strand Burner: a pressurized chamber that simulates chamber pressure of a rocket motor

Consists of 5 subsystems:

- Structures
  - Pressure System
  - Ignition
  - Burn-Away Copper Wires
  - Data Acquisition Software and Interfacing
- Include isometric cross section image here, label subsystems on image**



# Constants for moment calculations

$$\begin{aligned}
L_9 &= \frac{-r_o}{a} \left\{ \frac{1}{2} \ln \frac{r_o}{r_o} + \frac{1 - \left(\frac{r_o}{a}\right)^4}{4} \left[ 1 - \left(\frac{r_o}{a}\right)^4 \right] \right\} \\
L_{11} &= \frac{1}{64} \left\{ 1 + 4 \left(\frac{r_o}{a}\right)^2 - 5 \left(\frac{r_o}{a}\right)^4 - 4 \left(\frac{r_o}{a}\right)^2 \left[ 2 + \left(\frac{r_o}{a}\right)^2 \right] \ln \frac{a}{r_o} \right\} \\
L_{12} &= \frac{a}{14,400(a - r_o)} \left\{ 64 - 225 \left(\frac{r_o}{a}\right)^6 - 100 \left(\frac{r_o}{a}\right)^3 + 261 \left(\frac{r_o}{a}\right)^5 \right. \\
&\quad \left. + 60 \left(\frac{r_o}{a}\right)^3 \left[ 3 \left(\frac{r_o}{a}\right)^2 + 10 \right] \ln \frac{a}{r_o} \right\} \\
L_{13} &= \frac{a^2}{14,400(a - r_o)^2} \left\{ 25 - 128 \left(\frac{r_o}{a}\right)^3 + 225 \left(\frac{r_o}{a}\right)^2 - 25 \left(\frac{r_o}{a}\right)^4 \right. \\
&\quad \left. - 97 \left(\frac{r_o}{a}\right)^6 - 60 \left(\frac{r_o}{a}\right)^4 \left[ 5 + \left(\frac{r_o}{a}\right)^2 \right] \ln \frac{a}{r_o} \right\} \\
L_{14} &= \frac{1}{16} \left[ 1 - \left(\frac{r_o}{a}\right)^4 - 4 \left(\frac{r_o}{a}\right)^2 \ln \frac{a}{r_o} \right] \\
L_{15} &= \frac{a}{720(a - r_o)} \left[ 16 - 45 \left(\frac{r_o}{a}\right)^6 + 9 \left(\frac{r_o}{a}\right)^5 + 20 \left(\frac{r_o}{a}\right)^3 \left(1 + 3 \ln \frac{a}{r_o}\right) \right] \\
L_{16} &= \frac{a^2}{1440(a - r_o)^2} \left[ 15 - 64 \left(\frac{r_o}{a}\right)^2 + 90 \left(\frac{r_o}{a}\right)^4 - 6 \left(\frac{r_o}{a}\right)^6 - 5 \left(\frac{r_o}{a}\right)^4 \left(7 + 12 \ln \frac{a}{r_o}\right) \right] \\
L_{17} &= \frac{1}{4} \left\{ 1 - \frac{1 - \nu}{4} \left[ 1 - \left(\frac{r_o}{a}\right)^4 \right] - \left(\frac{r_o}{a}\right)^2 \left[ 1 + (1 + \nu) \ln \frac{a}{r_o} \right] \right\}
\end{aligned}$$

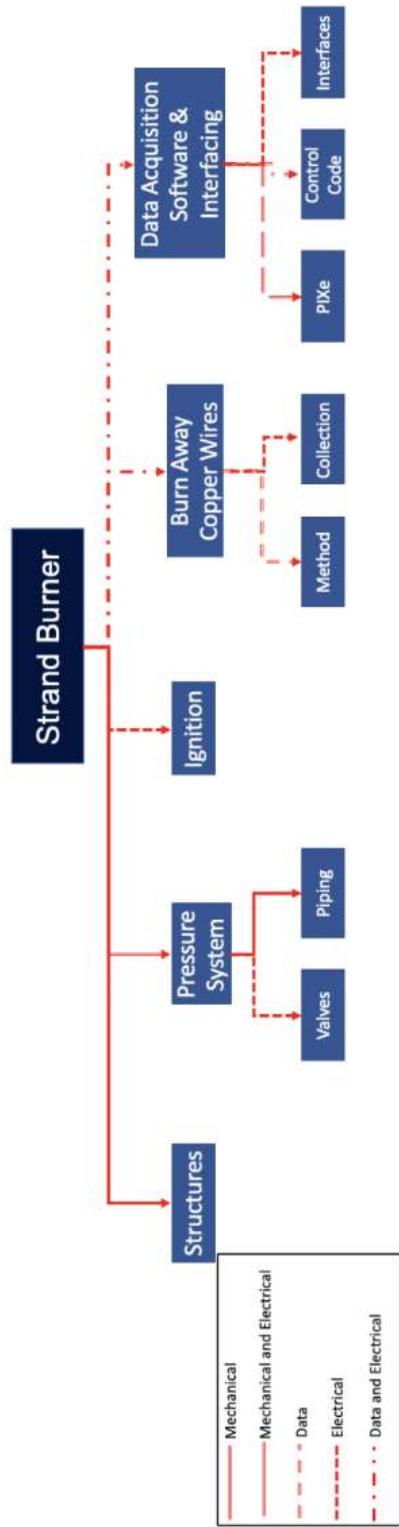
$$\begin{aligned}
C_1 &= \frac{1 + \nu}{2} \frac{b}{a} \ln \frac{a}{b} + \frac{1 - \nu}{4} \left( \frac{a}{b} - \frac{b}{a} \right) \\
C_2 &= \frac{1}{4} \left[ 1 - \left(\frac{b}{a}\right)^2 \left( 1 + 2 \ln \frac{a}{b} \right) \right] \\
C_3 &= \frac{b}{4a} \left\{ \left[ \left(\frac{b}{a}\right)^2 + 1 \right] \ln \frac{a}{b} + \left(\frac{b}{a}\right)^2 - 1 \right\} \\
C_4 &= \frac{1}{2} \left[ (1 + \nu) \frac{b}{a} + (1 - \nu) \frac{a}{b} \right] \\
C_5 &= \frac{1}{2} \left[ 1 - \left(\frac{b}{a}\right)^2 \right] \\
C_6 &= \frac{b}{4a} \left[ \left(\frac{b}{a}\right)^2 - 1 + 2 \ln \frac{a}{b} \right] \\
C_7 &= \frac{1}{2} (1 - \nu^2) \left( \frac{a}{b} - \frac{b}{a} \right) \\
C_8 &= \frac{1}{2} \left[ 1 + \nu + (1 - \nu) \left(\frac{b}{a}\right)^2 \right] \\
C_9 &= \frac{b}{a} \left\{ \frac{1 + \nu}{2} \ln \frac{a}{b} + \frac{1 - \nu}{4} \left[ 1 - \left(\frac{b}{a}\right)^2 \right] \right\}
\end{aligned}$$

# Subsystem Analysis- Pressure System

PID	Component	No Mitigation			With Mitigation			Severity Rating
		Failure Mode	Consequence of Failure	Likelihood	Severity	Rating	Mitigation	
REG-102	Panel Iso Regulator	Fail Full Open	Over-pressurization of circuit	2	4	8	Relief valve, pressure visibility, downstream isolation valves, proof to 1.5x MEOP	Loss of consumables
REG-105	Muscle Pressure Regulator	Fail Full Open	Over-pressurization of circuit	2	4	8	Pressure visibility, relief valve, proof to MEOP	Loss of consumable
MV-107	Muscle Pressure Vent	Misconfigured open	loss of consumable	4	2	8	Procedural control, training of personnel	Loss of consumable
MV-109	High Pressure Manual Vent	Misconfigured open	loss of consumable	4	2	8	Procedural control, training of personnel	Loss of consumable
MV-110	Manual Panel Iso	Misconfigured closed	unable to pressurize circuit, loss of time to reconfigure	4	1	4	Procedural control, training of personnel	minor loss of time
SV-111	Test Article Press		unwanted pressure in test article, possible loss of consumable	2	2	4	fail state is closed, procedural controls, upstream isolation valve	minor time loss
SV-113	Test Article Vent		Over-pressurization of test article	2	4	8	Proof to 1.5x MEOP, fail state is open	loss of consumable



# System Overview



# Hazardous Operation Assessment Definition

---

## Severity Ratings:

- 1 – Not a risk to system/personal/data and/or minor time loss
- 2 – Loss of consumables and/or minor to no injury to personnel and/or major time loss
- 3 – Damage to component and/or minor injury to personnel and/or loss of data
- 4 – Damage to system and/or loss of component and,/or severe injury to personnel
- 5 - Complete loss of article and/or loss of life

## Likelihood Ratings:

- 1 - Will most likely never occur in the life of the system
- 2 – Likely to occur once every 100+ uses of system
- 3 - Likely to occur once every 50+ uses of system
- 4 - Likely to occur once every 10+ uses of system
- 5 - Guaranteed to occur with every use of system

S	L	1	2	3	4	5
1	1	2	3	4	5	5
2	2	4	6	8	10	10
3	3	6	9	12	15	15
4	4	8	12	16	20	20
5	5	10	15	20	25	25

# Requirements

Requirement		Design Metric Benchmarks	Tools	Verification Method
Number	Title	Description		
1.1.1	Measurement Standards	Ability to measure chamber pressure, thrust and temperature	90% Confidence	Equipment error specifications
1.1.2	Measurement System Compatibility	Obtain data for 38 mm, 54 mm, 75 mm, 98 mm, 6 in, diameter motors, and strand burner	Be able to get data for all measurements	Strand Burner system and test materials
1.1.3	Detachability	The data acquisition cables must be detachable for storage.	Must be detachable	Personnel and appropriate wiring
1.1.4	Interface	Data acquisition system must interface with Pixe Test Cell 3.	Must successfully interface	Appropriate wiring
1.1.5	Ignition	Ignition system must set off propellant.	90% of the time	Strand Burner, ignition system, test propellant
1.1.6	Cost Cutting	Cut cost of Characterization	Cut by 15%	Previous testing cost data
				Verify by Spreadsheet



# Requirements – Cont.

Number	Title	Description	Design Metric Benchmarks		Tools	Verification Method
1.1.7	Strand Burner Integrity	The Ability for strand burner to safely function	10 years	Excel sheet, appropriate information required for calculations	Fatigue calculations	
1.1.8	Ignition Safety	Continuity light on while connected to ignition circuit	Lights up while there is continuity	Continuity light, supporting equipment	Electrical Continuity Test	
1.1.9	False Ignition Prevention	Prevent premature ignition	Lockout Keys prevent ignition	Lockout keys and switches, supporting equipment	Electrical Continuity Test	
1.1.10	Safety/Testing Procedures	Re-evaluation, adjustment, and re-approval of testing procedures	Safety standards	N/A	Safety, Dr. Bryner, RDL Safety Committee, Professor Gerrick	
1.1.11	Test Procedure Changes	Changes to test procedure must adhere to risk assessment	Risk assessment	N/A	Review, Safety, Dr. Bryner, RDL Safety	
1.1.12	Strand Burner Operations	No personnel allowed to be within the defined exclusion zone during pressurization	Exclusion zone map	Personnel	Follow proper SOPs and safety precautions	
1.1.13	Factor of Safety	Material withstands internal pressure	Factor of safety of 3 for the pipe and flanges and 1.2 for bolts	Strand burner and pressure subsystem	Theoretical calculations and pressureize to the highest pressure	
1.1.14	Pressurization Circuit Proof	Circuit will be proofed to the ASTM B31.3 Standard	Hydrostatic proof test to 1.5x MEOP	Hydrostatic Pump	Hydrostatic proof test	

# Trade Studies

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Following alternative were considered while design the current system:

(also include picture that's in poster)

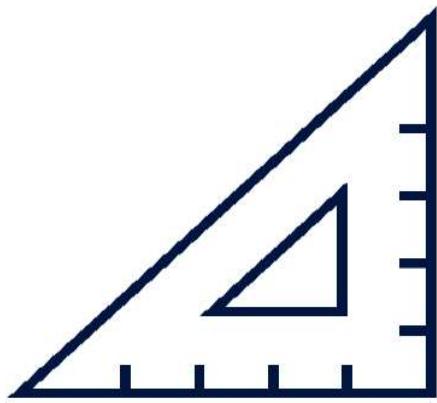
- Closed chamber for maintaining constant pressure

- Discrete voltage/capacitance measurement for data

- Laser interferometer (include what we learnt )



# Structures



# Subsystem Governing Equation – Structures

$$n = \frac{S_y}{\sigma}$$

*Equation 2: Factor of Safety for Pipe [4]*

$$n_p = \frac{S_p A_t}{C_k P + F_i}$$

*Equation 3: Yielding Factor of Safety for Bolts [4]*

$$T_2 = \frac{\Delta U}{m C_p} + T_1$$

*Equation 5: Energy Release from the combustion of the Propellant [5]*

n = factor of safety

$A_t$  = tensile area

$C_v$  = specific heat capacity at constant volume

$C_k$  = stiffness of frustum

P = external tensile load

$F_i$  = preload

$\Delta U$  = internal energy

m = mass of argon gas and propellant gasses

$T_2$  = final temperature in chamber

$T_1$  = initial temperature of chamber before burning of propellant



# Subsystem Trade Studies – Structures

Criteria	Weight	Scale	Steel 4140	Steel 6150	Aluminum 6061
Cost	20	9 (Best)	5	1	9
Material Strength	30	5	9	9	9
Availability	30	1 (Worst)	9	1	1
Machinability	15		5	1	9
Total			<b>715</b>	<b>335</b>	<b>615</b>





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# Pressure System



# Subsystem Trade Studies – Pressure System

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Trade Study Results: Ball Valve

- Excellent durability and functionality
- Good compromise between speed of actuation and high-pressure capability
- Cheaper than other trade options



# Subsystem Summary- Pressure System

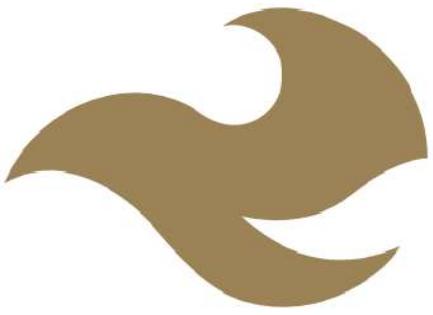
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It integrates into the system by connecting the:

- Pressurant gas to the valves and strand burner
- Valves to the data and control cables

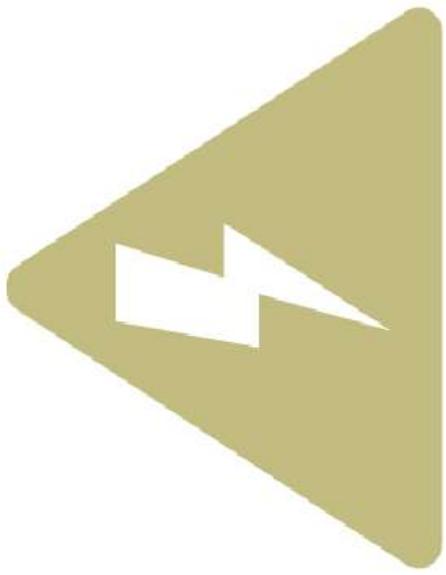
Number	Title	Value	Analysis Value	MOC	Meets Requirement S
1.1.13	Factor of Safety	≥ 3	3	Hydrostatic Testing	TBD





# Ignition





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# Burn Away Copper Wires



# Subsystem Trade Studies – Burn-Away Copper Wires

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Trade Study Results: Discrete Voltage Measurement

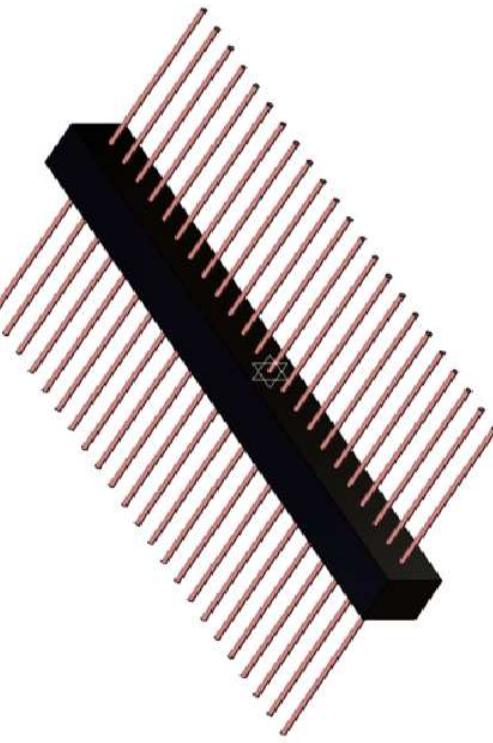
- Easiest to standardize
- Easiest to implement

• Reduces recurring requests the most

Number	Title	Value	Analysis Value	MOC	Meets Requirements
1.1.1	Measurement standards	≥ 90% confidence	TBD	Error Analysis	TBD
1.1.1	Measurement Standards	100 Hz	TBD	Run Data Acquisition Test	TBD



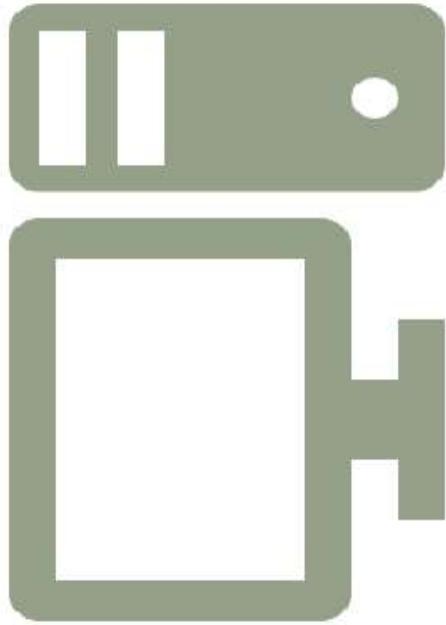
# Subsystem Definition – Burn-Away Copper Wires



Discrete Voltage Measurement Method is used to collect the burn rate data

Copper wires are imbedded inside the propellant

As the propellant burns the wires melt breaking the circuit



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# Data Acquisition Software and Interfacing



# Subsystem Summary- Data Acquisition Software and Interfacing

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Requirements Number	Title	Method of Characterization
1.1.3 & 1.1.4	DAC	Check cables interface with system and are detachable and storable
1.1.8 & 1.1.9	Ignition	Lock out key prevents ignition and light turns on with continuity



# Subsystem Analysis- Data Acquisition Software and Interfacing

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The team will perform a dry run test setup:

- Connection of the data acquisition cables interfacing the strand burner to the PXIe will be checked
- PXIe will be connected to the ignition system to check for continuity



# Full Bill of Materials

ITEM	QUANTITY	PRICE	LINK	SHIPPING	TAX	TOTAL PRICE
<b>Hydrostatic test</b>						
Water pump	1	\$ 6,513.00	<a href="https://www.s">https://www.s</a>	\$ 100.00	\$ 521.04	\$ 7,134.04
Water pump hose	1	\$ 270.00	<a href="https://www.s">https://www.s</a>	\$ 50.00	\$ 21.60	\$ 341.60
<b>Strand Burner</b>						\$ 1,002.52
6" diameter .020 steel pipe	1	\$ 328.25	<a href="https://www.s">https://www.s</a>	\$ 23.60	\$ 26.26	\$ 378.11
steel round bar (.140 hot rolled)	1 (8 in length)	\$ 221.97	<a href="https://www.s">https://www.s</a>	\$ 55.00	\$ 17.76	\$ 294.73
mounting brackets (A36 2x2x20)	1	\$ 127.46	<a href="https://www.s">https://www.s</a>	\$ 50.00	\$ 10.20	\$ 187.66
Pipefitting at a shop						\$ -
tube sealant tape	1	\$ 112.99	<a href="https://www.s">https://www.s</a>	\$ 20.00	\$ 9.04	\$ 142.03
plumbing sealant tape	1	\$ 87.79	<a href="https://www.s">https://www.s</a>	\$ 20.00	\$ 7.02	\$ 114.81
<b>Wiring</b>						\$ 1,169.82
wire pass through	1	\$ 800.00		\$ 20.00	\$ 64.00	\$ 884.00
railing	2	\$ 36.62	<a href="https://www.s">https://www.s</a>	\$ 20.00	\$ 2.93	\$ 59.55
nema-4 rated box	1	\$ 180.99	<a href="https://www.s">https://www.s</a>	\$ 20.00	\$ 15.28	\$ 226.27
<b>Plumbing</b>						\$ 2,334.57
1/4" pipes	3	\$ 145.38	<a href="https://www.s">https://www.s</a>	\$ 20.00	\$ 11.63	\$ 177.01
Pipe joints?	1	\$ 200.00		\$ 20.00	\$ 16.00	\$ 236.00
regulator 1 (7 k in 0.15k out)	1	\$ 275.00	<a href="https://www.s">https://www.s</a>	\$ 20.00	\$ 22.00	\$ 317.00
regulator 2 (6 k in 0.400 out)	1	\$ 275.00	<a href="https://www.s">https://www.s</a>	\$ 20.00	\$ 22.00	\$ 317.00
argon bottles	1	\$ 200.00		\$ 50.00	\$ 16.00	\$ 266.00
valve switches	2	\$ 20.00		\$ 10.00	\$ 1.60	\$ 31.60
relief valves	1	\$ 44.10	<a href="https://www.s">https://www.s</a>	\$ 20.00	\$ 3.53	\$ 67.63
burst disk	1	\$ 8.79	<a href="https://www.s">https://www.s</a>	\$ 10.00	\$ 0.70	\$ 19.49
feet hole	2	\$ 165.18	<a href="https://www.s">https://www.s</a>	\$ 20.00	\$ 13.21	\$ 196.79
Panel (3x3 6065 alum)	1	\$ 94.51	<a href="https://www.s">https://www.s</a>	\$ 20.00	\$ 7.56	\$ 122.07
3000 ns Pressure Transducer	1	\$ 301.30	<a href="https://www.s">https://www.s</a>	\$ 20.00	\$ 24.10	\$ 345.40
wheels for the panel (300 lb hand cart)	1	\$ 44.98	<a href="https://www.s">https://www.s</a>	\$ 10.00	\$ 3.60	\$ 58.58
<b>Propellant Samples</b>						\$ 206.80
propellant lig	1	\$ 39.44	<a href="https://www.s">https://www.s</a>	\$ 10.00	\$ 3.16	\$ 52.60
thin copper wire .02 mm	1	\$ 10.00	<a href="https://www.s">https://www.s</a>	\$ 10.00	\$ 0.80	\$ 20.80
igniters	1	\$ 55.00	<a href="https://www.s">https://www.s</a>	\$ 10.00	\$ 4.40	\$ 69.40
ig-sense wire	1	\$ 50.00		\$ 10.00	\$ 4.00	\$ 64.00



▷ Presentations	92 days	Fri 1/28/22	Fri 4/29/22		3 days
Delta PDR	1 day	Fri 1/28/22	Fri 1/28/22	18	94 days
CDR	1 day	Fri 2/4/22	Fri 2/4/22	20	48FS-1 d 0 days
TRR	1 day	Tue 3/22/22	Tue 3/22/22	133,122	135,128, 0 days
Final Technical Symposium	1 day	Wed 4/27/22	Wed 4/27/22	152	147,6 0 days
		Fri 4/29/22	Fri 4/29/22	5	0 days
▷ Design	22 days	Wed 1/12/22	Thu 2/3/22		88 days
▷ Control Code (Labview)	52 days	Fri 1/14/22	Mon 3/7/22		56 days
▷ Procurement	21 days	Fri 2/4/22	Thu 2/24/22	3FS-1 day,35,30,25	123 0 days
▷ Fabrication	12 days	Fri 2/25/22	Tue 3/8/22		11 days
Hydrostatic Test	2 days	Wed 3/9/22	Thu 3/10/22	116,49,54,59	135,4,13 11 days
Test Plan	30 days	Fri 2/25/22	Sat 3/26/22	48	135 0 days
▷ Test Readiness Review	9 days	Sun 3/13/22	Mon 3/21/22		0 days
▷ Testing	13 days	Sun 3/27/22	Fri 4/8/22	122,46,123,4	0 days
propellant production	5 days	Sun 3/27/22	Thu 3/31/22	4	137,143 0 days
test 1-3	2 days	Fri 4/1/22	Sat 4/2/22	136,4,128,122	138 7 days
test 4-6	2 days	Sun 4/3/22	Mon 4/4/22	137,4,128,122	139 7 days
test 7-9	2 days	Tue 4/5/22	Wed 4/6/22	138,4,128,122	140 7 days
System Verification	2 days	Thu 4/7/22	Fri 4/8/22	139,4	146FF+3 7 days
▷ Final Report	32 days	Fri 4/1/22	Mon 5/2/22		0 days
▷ Final Technical Review	8 days	Tue 4/19/22	Tue 4/26/22		0 days
Testing Manual	39 days	Fri 3/25/22	Mon 5/2/22		0 days



# Nema 4 Rated Electrical Box

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# Overview of Verification Tests

<u>Leak Check:</u>	<u>Proof Test:</u>	<u>Ignition Test:</u>	<u>Dry Run:</u>
<ul style="list-style-type: none"><li>• Verification of seals and torquing</li><li>• Pressure set to ~100 psi but no greater than 150 psi.</li><li>• Performed with Argon</li><li>• Panel, Burner, Full system</li></ul>	<ul style="list-style-type: none"><li>• Verification of pressure 1.5x or 3x above rating</li><li>• Performed with water and hydrostatic pump</li><li>• Pressures of 4500 psi for pressure system and 3000 psi for burner</li></ul>	<ul style="list-style-type: none"><li>• Perform 3* tests</li><li>• Place E-match into propellant “blob”</li><li>• Verify E-match ignites propellant, ignites and ignition circuit work correctly</li></ul>	<ul style="list-style-type: none"><li>• Go through procedures as if doing actual test</li></ul>

\* Subject to change



# Subsystem Trade Studies – Structures

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Trade Study Results:

Pipe: ASTM A500 Grade B Steel

- High material strength and availability compared to other options
- Easier to machine
- Lowest cost compared to other options

Caps: 12L14 Cold Rolled

- High material strength
- Easier to machine



# Cost Breakdown by High-Cost Item

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Item	Cost	Use/Purpose
Wire Passthrough	\$884	Allows data transfer
Regulators	\$634	Regulates the pressure between the argon bottle and the pressure system
Strand Burner Body	\$380	Stand burner material
Flex Hose	\$376.79	Helps gas travel through sharp turns
Pressure Transducer	\$301.30	Measures the internal pressure
Pipe Threading	\$280	No capabilities on campus

