

Central Force Motion

$$\mu \ddot{\mathbf{r}} = \mu r \dot{\theta}^2 + f(r)$$

$$\mu (r\ddot{\theta} + 2\dot{r}\dot{\theta}) = 0 \longrightarrow \vec{L} = \vec{\mu} \vec{r} \times \vec{v} = l \hat{k}$$

constant

$$l = \mu r^2 \dot{\theta}$$

$$\frac{1}{2} \mu \dot{r}^2 + \underbrace{\frac{l^2}{2\mu r^2}} + U(r) = E : \text{constant}$$

$$\mu \equiv \frac{mM}{M+m}$$

$$\equiv U_{\text{eff}}(f) \left\{ -\frac{du}{dr} = f \right\}$$

Kepler Problem

$$U(r) = -\frac{GMm}{r}$$

$$\frac{d\theta}{dt} = \frac{l}{\mu r^2}$$

$$\frac{dr}{dt} = \sqrt{\frac{2E}{\mu} - \frac{l^2}{\mu^2 r^2} - \frac{2U}{\mu}}$$

$$\frac{dr}{d\theta} = g(r)$$

$$\rightarrow \int \frac{dr}{g(r)} = \int d\theta$$

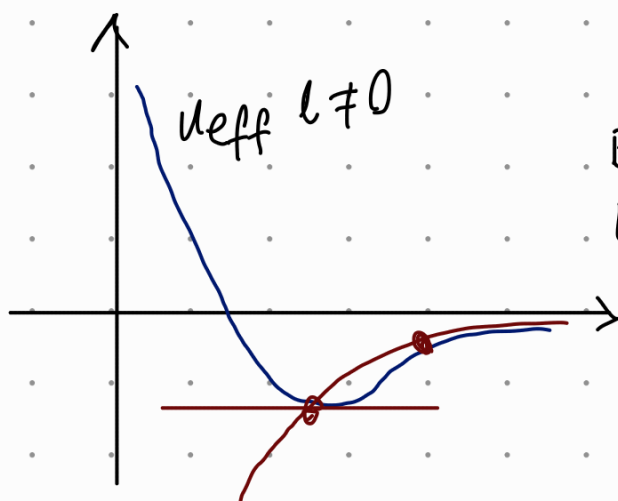
$$r = \frac{r_0}{1 - \epsilon \cos \theta}$$

$$r_0 = \frac{l^2}{\mu c}$$

$$\epsilon = \sqrt{1 + \frac{E l^2}{\mu c^2}}$$

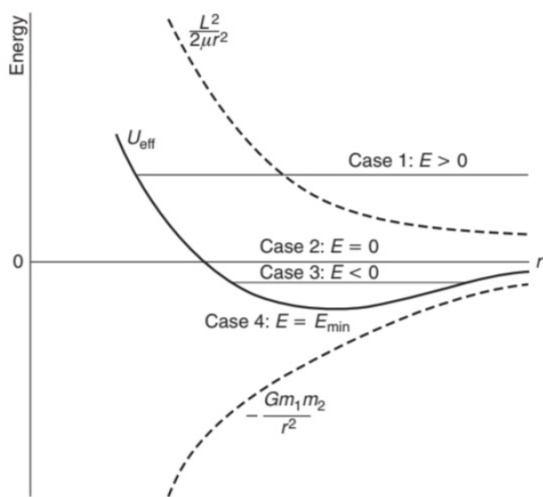
↓
eccentricity

$$c = GmM \quad \epsilon > 0$$



Bound orbits must have $E < 0$

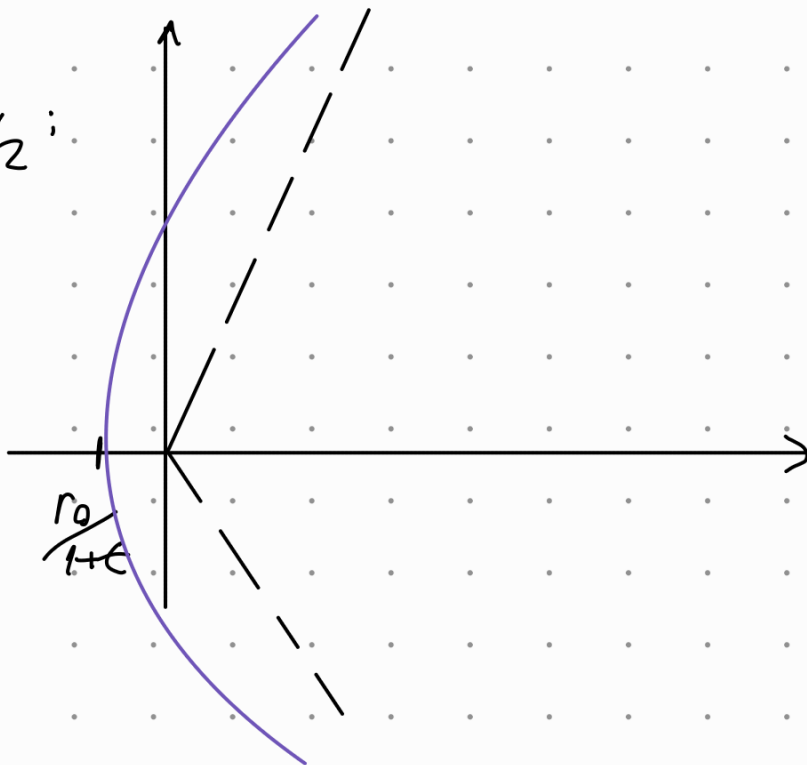
Unbound orbits have $E \geq 0$



$$E = 0 \Rightarrow r = r_0$$

• $E = 2$ $\cos \theta = 1/2$;

see there is a bound to θ when $e > 1$



Hyperbola?

• $e = 1/2$
 $x = r \cos \theta$

$y = r \sin \theta$

$$\sqrt{x^2 + y^2} = \frac{r_0 \sqrt{x^2 + y^2}}{\sqrt{x^2 + y^2} - e x}$$

(...)

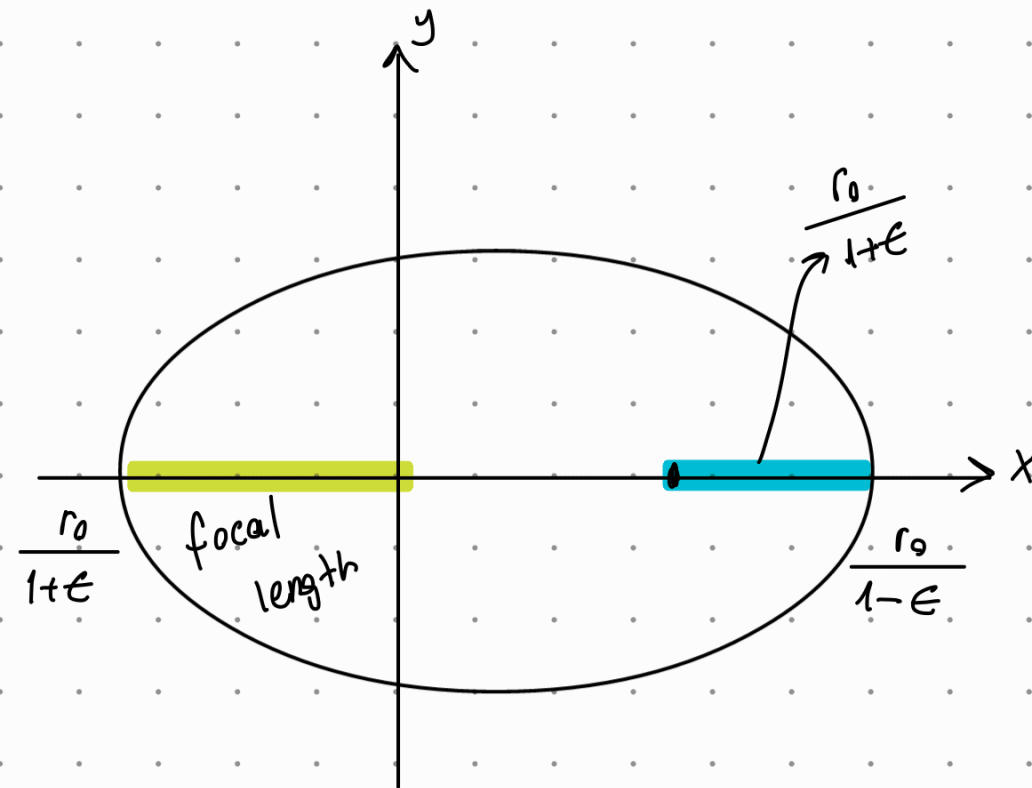
$$(1 - e^2)x^2 - 2r_0 e x + y^2 = r_0^2$$

quadratic curve

• hyperbola
 • parabola

"Show this is a circle when $\epsilon = 0$."

"Scattering Solutions"



$$\frac{r_0}{1+\epsilon} + \frac{r_0}{1-\epsilon} = \frac{2 \cdot r_0}{1-\epsilon^2} = \frac{\frac{2l^2}{\mu c}}{\frac{-2El^2}{\mu c^2}} = \frac{c}{-E}$$

major axis

$$1 - \left(1 + \frac{2El^2}{\mu c^2} \right) = \frac{2El^2}{\mu c^2}$$

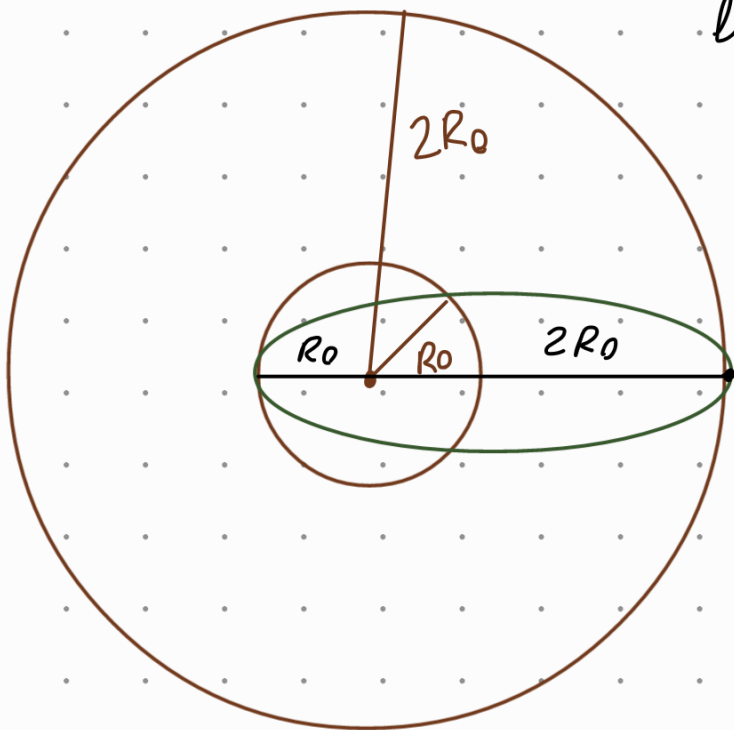
- Say you want to change your circular orbit for another one.
→ What happens?

Dünyadan Kurtulmak

$$\frac{1}{2} \mu \dot{r}^2 - \frac{GMm}{R_e} = E > 0$$

$$(r^2)_{\min} = \frac{2GMm}{\mu R_e} = \frac{2G(M+m)}{R_e} = \frac{2GM_e}{R_e}$$

"How did Voyager get out of solar system?"
→ Uranus, Jupiter helped (SlingShot)



$$l = \mu R_0^2 \dot{\theta} = -\frac{1}{2} \frac{\mu C^2}{l^2}$$

$$E = \frac{l^2}{2\mu R_0^2} - \frac{C}{R_0} = \frac{\mu C^2}{2l^2} - \frac{C^2 \mu}{l^2}$$

$$-\frac{l^2}{\mu R_0^2} + \frac{C}{R} = 0$$

$$R_0 = \frac{l^2}{\mu C}$$

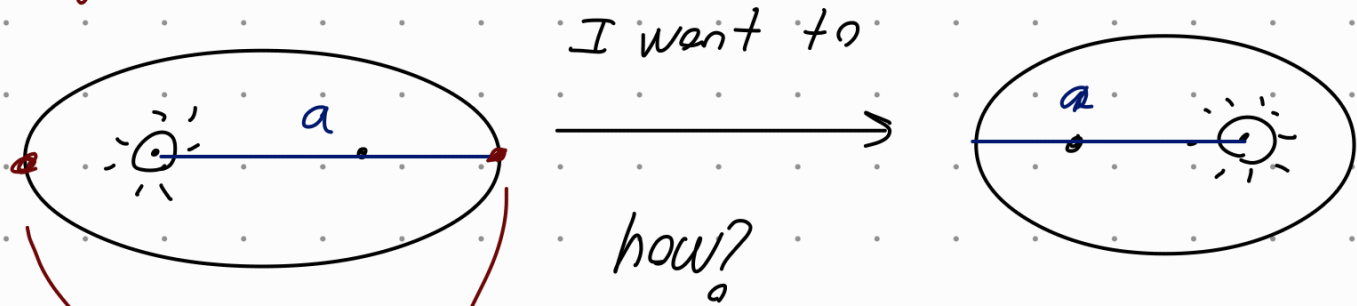
"effective shock"

$$= -\frac{1}{2} \frac{C}{R_0}$$

$$\frac{r_0}{1+\epsilon} = 2R_0 \quad r_0 = (1-\epsilon)2R_0 \quad \frac{1+\epsilon}{1-\epsilon} = 2$$

"What to do for low ϵ ?"

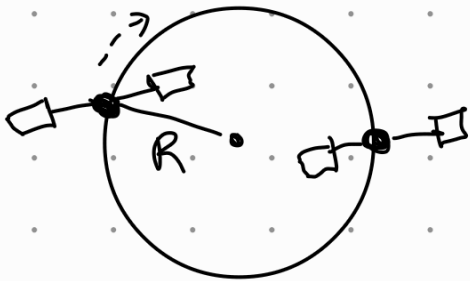
→ Question



need a circle passing from this points.

→ Question

"Can you send help to the other satellite in the same circular orbit"



→ Slow down and wait up for the other one to catch up to you.

→ Question How to fall down to earth?

$W \swarrow$ $L \swarrow$ U_{eff} gets shallow,
you fall down.

