

FORWARD FUSELAGE II - COCKPIT BOTTOM - INTERNAL - INSPECTION/CHECK

EFFECTIVITY: ALL

1. General

- A. This section gives the procedures to do the visual inspection of forward fuselage II - cockpit bottom for general condition.
- B. Related Zones: 121/122/123/124.
- C. Zone Boundaries: STA X = 2364.0 thru STA X = 4154.5.
- D. The procedures in this section are given in the sequence below. The tasks identified with (♦) are part of the Scheduled Maintenance Requirements Document (SMRD).

TASK NUMBER	DESCRIPTION	EFFECTIVITY
05-20-11-200-801-A ♦	FORWARD FUSELAGE II - COCKPIT BOTTOM - INTERNAL GENERAL VISUAL INSPECTION	ALL

TASK 05-20-11-200-801-A

EFFECTIVITY: ALL

2. FORWARD FUSELAGE II - COCKPIT BOTTOM - INTERNAL GENERAL VISUAL INSPECTION

A. General

- (1) This procedure obeys the EWIS ICA requirement.
- (2) This task gives instructions to do SRMD Zonal Task 53-Z123-214-001-A00.
- (3) You must do the internal general visual inspection (GVI) of forward fuselage II, at zones 121/122/123/124, within a touching distance of the items examined.
- (4) The function of the internal general visual inspection (GVI) is to find damage, failure, or irregular conditions that can be easily seen.
- (5) Apart from any specific targets defined in the Zonal Inspection task, a General Visual Inspection (GVI) must be conducted of the entire zone specified and includes all installations, components and structures. Refer to AMM Introduction for zonal inspection criteria.

B. References

REFERENCE	DESIGNATION
AMM MPP 06-41-01/100	-
AMM TASK 12-23-00-600-801-A/300	-
AMM TASK 25-81-01-000-801-A/400	-
AMM TASK 25-81-01-000-802-A/400	-
AMM TASK 25-81-01-400-801-A/400	-
AMM TASK 25-81-01-400-802-A/400	-
AMM TASK 28-41-00-200-801-A/600	-
AMM TASK 53-01-01-000-801-A/400	COCKPIT FLOOR PANELS - REMOVAL
AMM TASK 53-01-01-400-801-A/400	COCKPIT FLOOR PANELS - INSTALLATION
WM 20-22-00	-

C. Zones and Accesses

ZONE	PANEL/DOOR	LOCATION
123	123BL	Forward fuselage II
221	221AF	Cockpit floor panel
221	221BF	Cockpit floor panel
221	221CF	Cockpit floor panel
221	221DF	Cockpit floor panel
221	221EF	Cockpit floor panel
221	221FF	Cockpit floor panel
221	221GF	Cockpit floor panel
222	222AF	Cockpit floor panel

(Continued)

ZONE	PANEL/DOOR	LOCATION
222	222BF	Cockpit floor panel
222	222CF	Cockpit floor panel
222	222DF	Cockpit floor panel
222	222EF	Cockpit floor panel
222	222FF	Cockpit floor panel
222	222GF	Cockpit floor panel

D. Tools and Equipment

ITEM	DESCRIPTION	PURPOSE	QTY
Standard	Ladder	To get access to the inspection area	
Standard	Flashlight	To make the inspection of the area easier	

E. Auxiliary Items

Not Applicable

F. Consumable Materials

Not Applicable

G. Expandable Parts

Not Applicable

H. Persons Recommended

QTY	FUNCTION	PLACE
1	Does the task	Forward fuselage II - cockpit bottom

I. Preparation (Figure 601) (Figure 602)

SUBTASK 841-002-A

- (1) Open access door 123BL (AMM MPP 06-41-01/100) (Figure 601).
- (2) Remove cockpit floor panels 222AF, 221AF, 221BF, 221CF, 221DF, 221EF, 221FF, 222CF, 222EF, 222GF, 222DF, 222FF, 222BF, and 221GF (AMM TASK 53-01-01-000-801-A/400) (Figure 602).
- (3) Remove the thermoacoustic insulation pads, at zones 123/124 (AMM TASK 25-81-01-000-801-A/400) or (AMM TASK 25-81-01-000-802-A/400) as applicable.

J. Internal General Visual Inspection (Figure 603) (Figure 604) (Figure 605)

SUBTASK 212-002-A

- (1) Examine the interior of the zone for loose rivets, skin deformation, nicks, cracks, dents, sealant poor condition, scratches, gouges, creases, erosion, corrosion, painting condition, deteriorated protective treatment, and foreign objects.
- (2) Examine the interior of the fuselage skin panel at stringers, frames/bulkheads, pressure bulkhead, and longitudinal and circumferential splices for loose rivets, cracks,

skin deformation, painting, erosion, corrosion, and deteriorated protective treatment (Figure 603).

- (3) Examine the interior of the cockpit floor beams and columns for loose rivets, cracks, deformation, distortion, wrinkles, erosion, corrosion, and deteriorated protective treatment (Figure 603).
- (4) Examine drains for signs of corrosion. Also do a check for blockage and correct operation (AMM TASK 12-23-00-600-801-A/300).
- (5) Examine the interior of the zone for leakage on the skin, signs of friction, wear, distortion, tension, broken strands, and corrosion of the control cables from the rudder, elevator, and control wheel to the forward quadrant (Figure 604).
- (6) Examine the interior of the zone for bad condition of the structure of the aileron and elevator disconnect system, and pedal assembly (Figure 604).
- (7) Examine the mechanical gust lock system and stick pusher actuator mechanical linkage for loose rivets, signs of friction, scratches, gouges, deformation, distortion, erosion, and corrosion at the connecting rod and tension spring (Figure 605).
- (8) Examine the electromechanical gust lock system and stick pusher actuator mechanical linkage for loose rivets, signs of friction, scratches, gouges, deformation, distortion, erosion, and corrosion at the spring assembly (Figure 606).
- (9) Examine the interior of the zone for signs of friction, tension, broken strands, corrosion of the pressurization duct, and signs of overheating caused by leakage at the connection of hydraulic lines and air conditioning ducts and outlets (Figure 607).
- (10) Examine the EWIS components for signs of damage, adequate installation, chafing and general condition. Refer to WM 20-22-00.

K. Follow-on

SUBTASK 842-002-A

- (1) Install thermoacoustic insulation pads, at zones 121/122/123/124 (AMM TASK 25-81-01-400-801-A/400) or (AMM TASK 25-81-01-400-802-A/400) as applicable.
- (2) Do an inspection on the fuel quantity indication harness (AMM TASK 28-41-00-200-801-A/600).

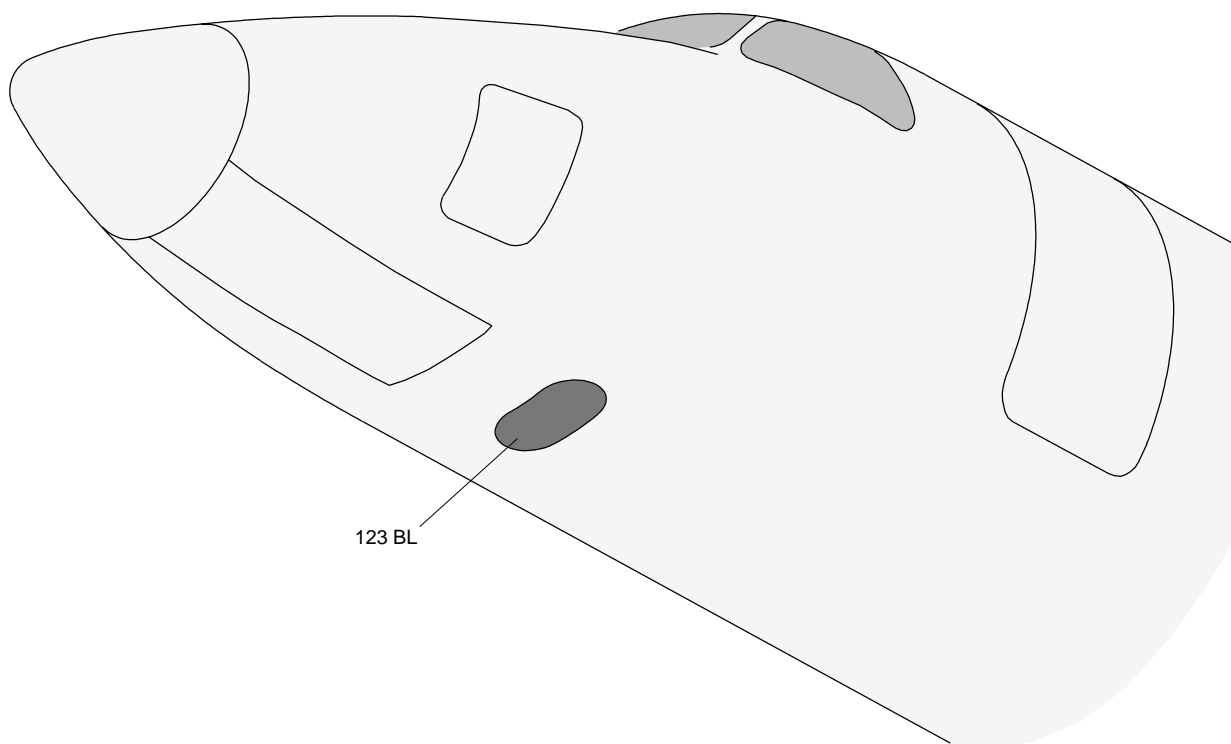
NOTE: The inspection of fuel quantity indication harness is a part of Critical Design Configuration Control Limitations (CDCCL) in the Airworthiness Limitations of the Aircraft Maintenance Program.

- (3) Install cockpit floor panels 221GF, 222BF, 222FF, 222DF, 222GF, 222EF, 222CF, 221FF, 221EF, 221DF, 221CF, 221BF, 221AF, and 222AF (AMM TASK 53-01-01-400-801-A/400).
- (4) Close access door 123BL (AMM MPP 06-41-01/100).

EFFECTIVITY: ALL

Forward Fuselage II - Cockpit Bottom - Cockpit - Access Door

Figure 601

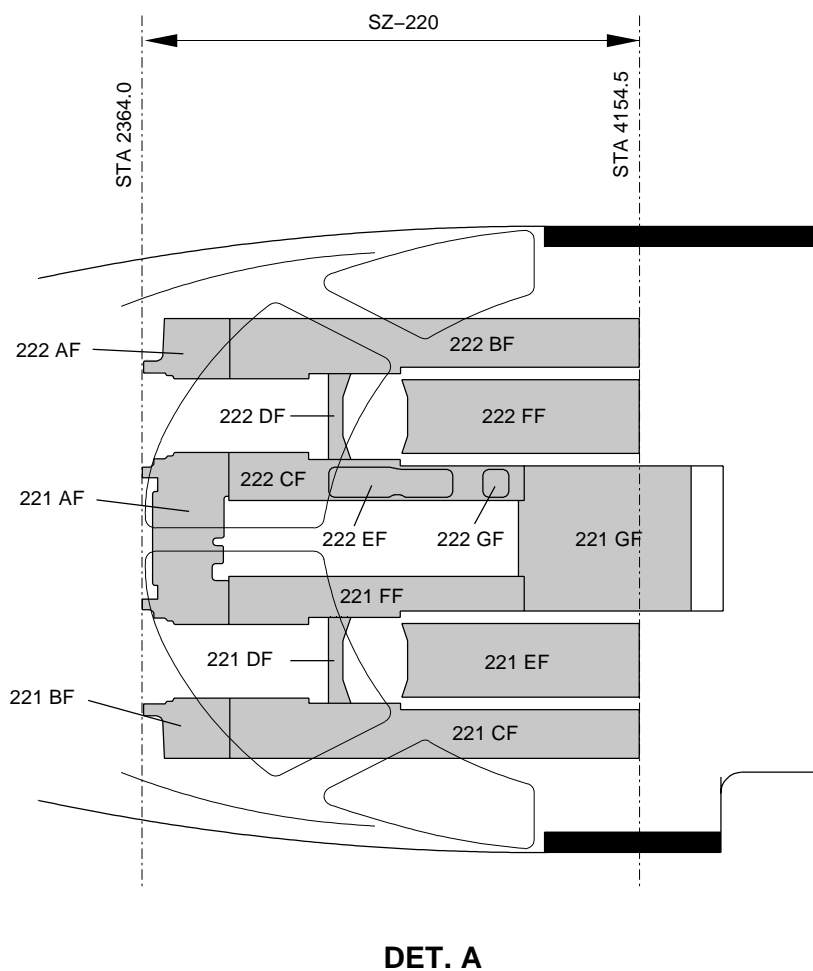
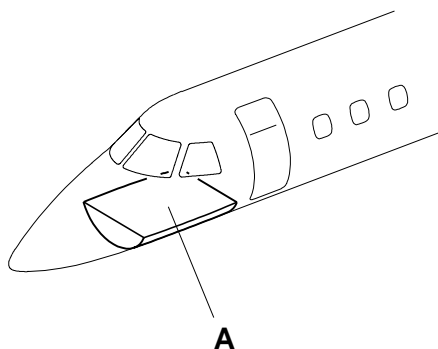


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EFFECTIVITY: ALL

Forward Fuselage II - Cockpit Bottom - Access Floor Panels

Figure 602

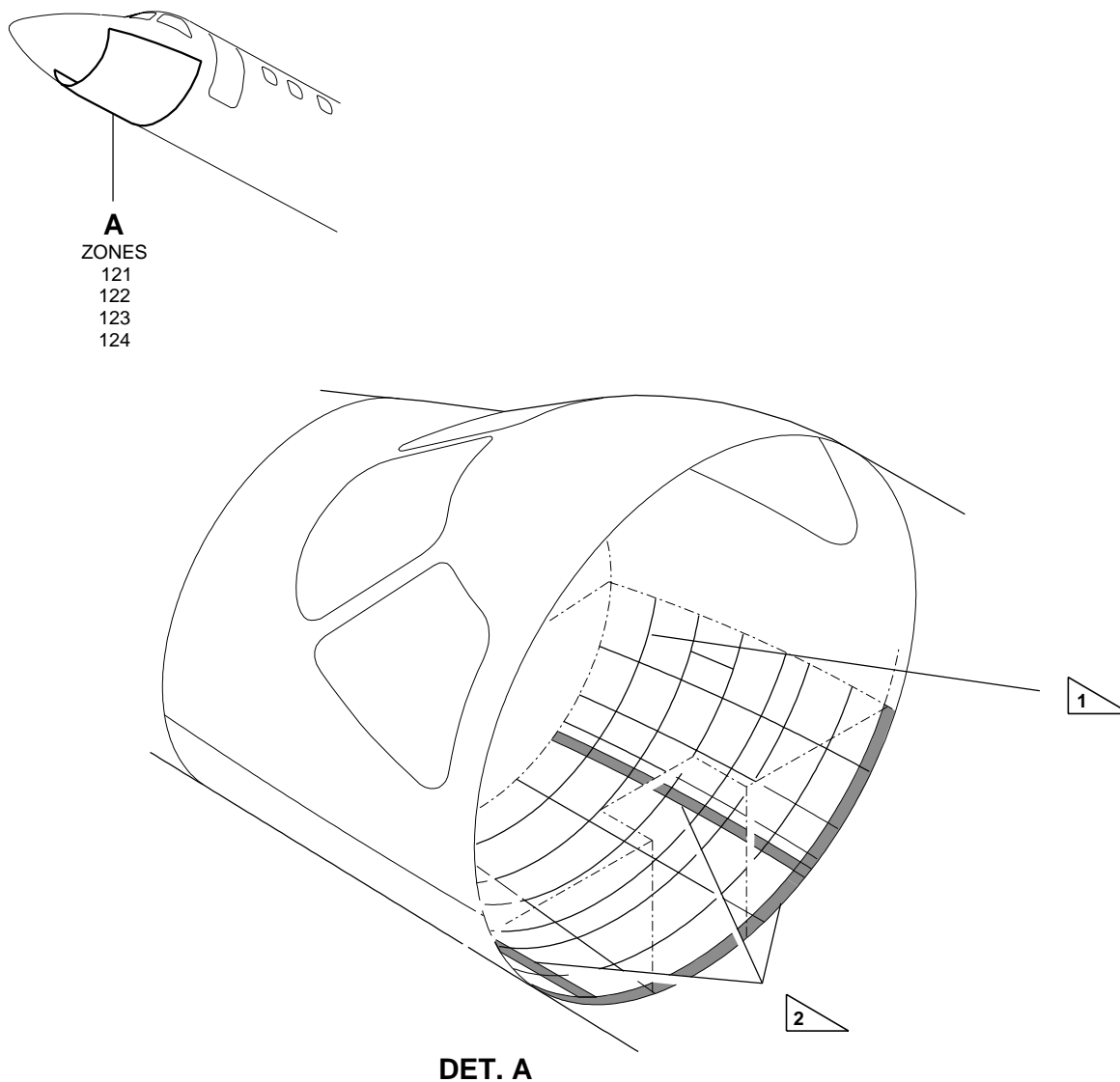


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EFFECTIVITY: ALL

Forward Fuselage II - Cockpit Bottom - Internal General Visual Inspection

Figure 603 - Sheet 1



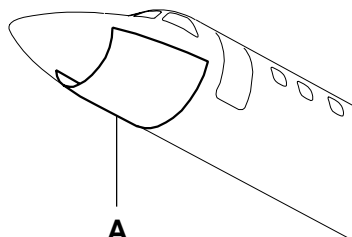
- 1 INTERNALLY INSPECT THE FUSELAGE SKIN PANEL AT STRINGERS, FRAMES/BULKHEADS, AND PRESSURE BULKHEAD.
- 2 INTERNALLY INSPECT THE FUSELAGE SKIN LONGITUDINAL AND CIRCUMFERENTIAL SPLICES.

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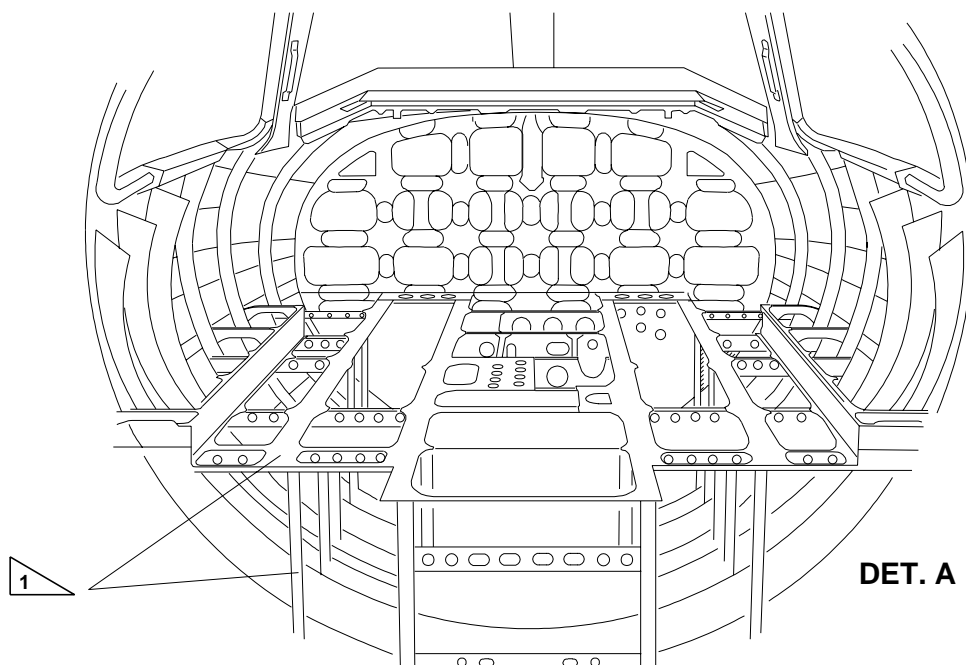
EFFECTIVITY: ALL

Forward Fuselage II - Cockpit Bottom - Internal General Visual Inspection

Figure 603 - Sheet 2



A
ZONES
121
122
123
124



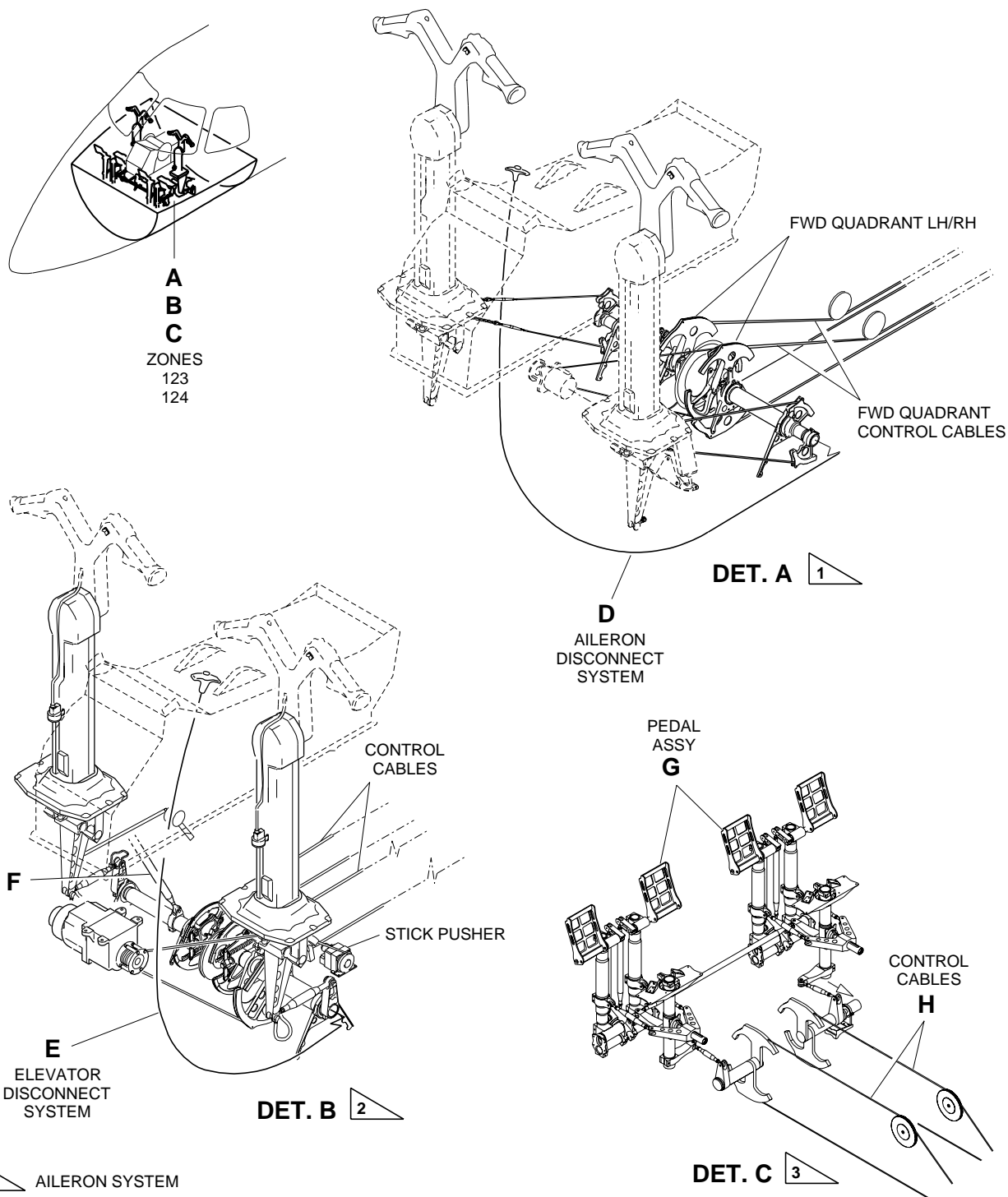
1 INTERNALLY INSPECT THE COCKPIT FLOOR BEAMS AND COLUMNS.

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EFFECTIVITY: ALL

Forward Fuselage II - Cockpit Bottom - Aileron, Elevator, and Rudder Systems

Figure 604 - Sheet 1

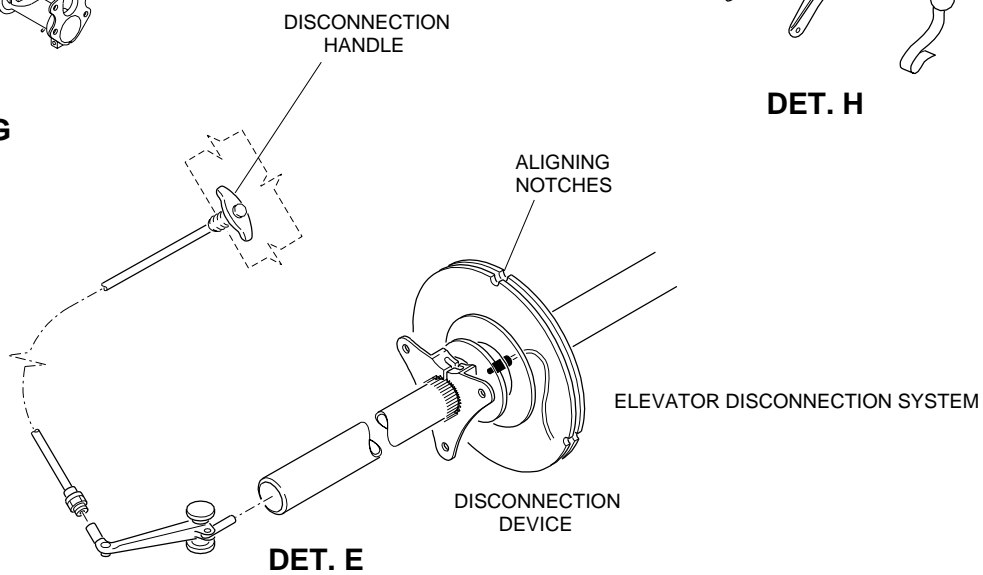
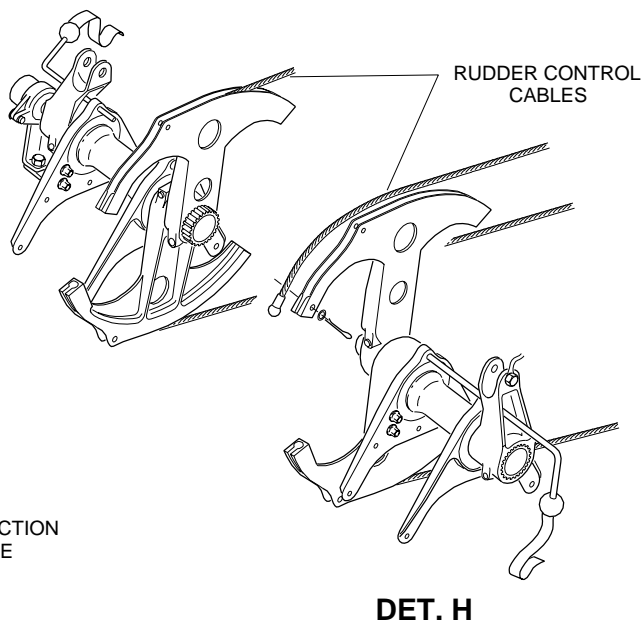
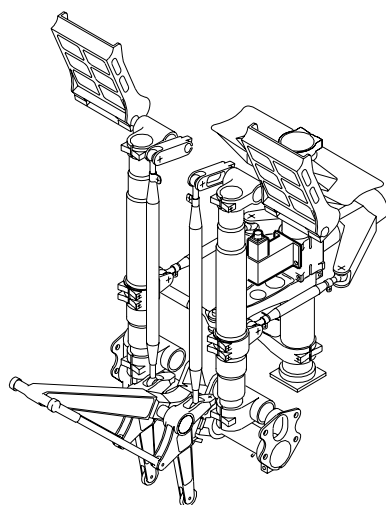
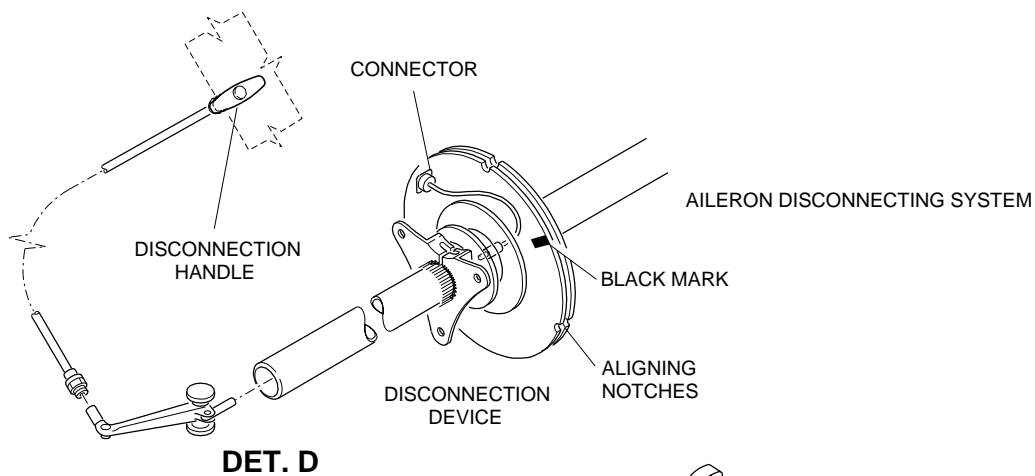


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EFFECTIVITY: ALL

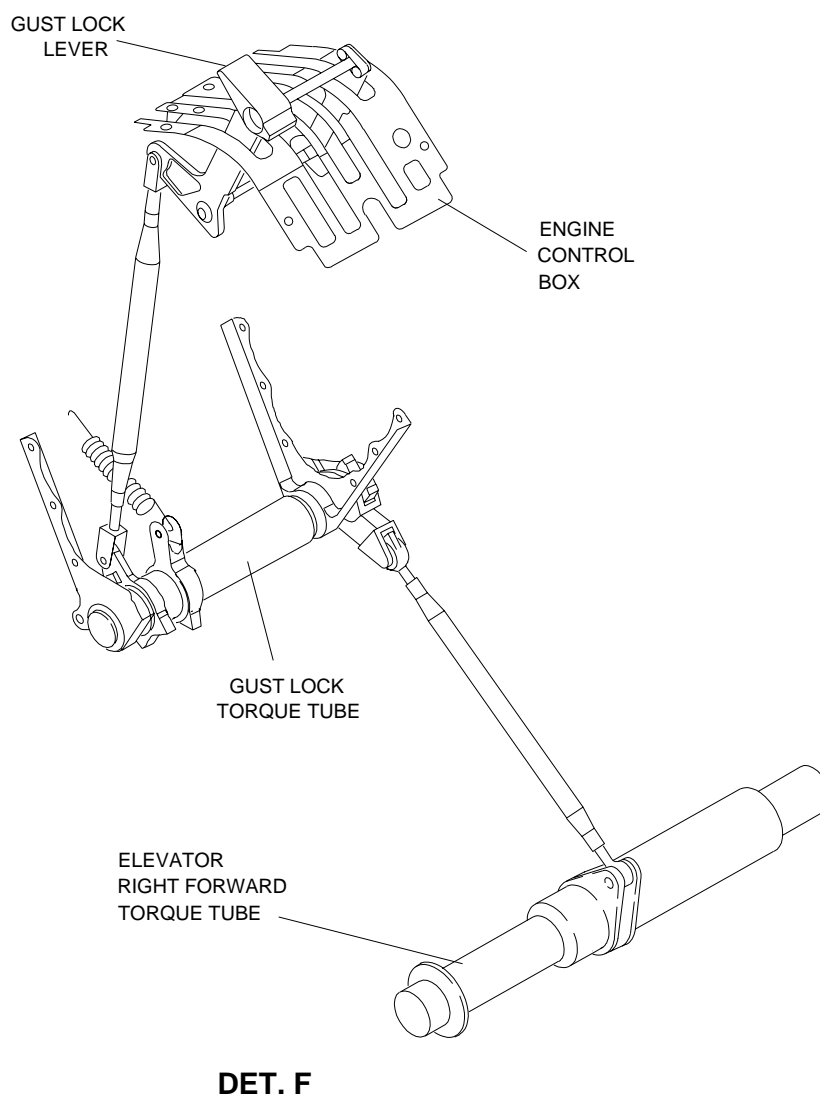
Forward Fuselage II - Cockpit Bottom - Aileron, Elevator, and Rudder Systems

Figure 604 - Sheet 2



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EFFECTIVITY: PRE-MOD. S.B. 145-27-0075, AIRCRAFT WITH MECHANICAL GUST LOCK SYSTEM
Forward Fuselage II - Cockpit Bottom - Mechanical Gust Lock
Figure 605

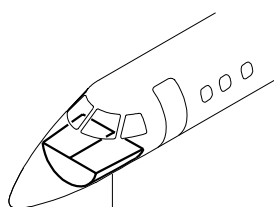


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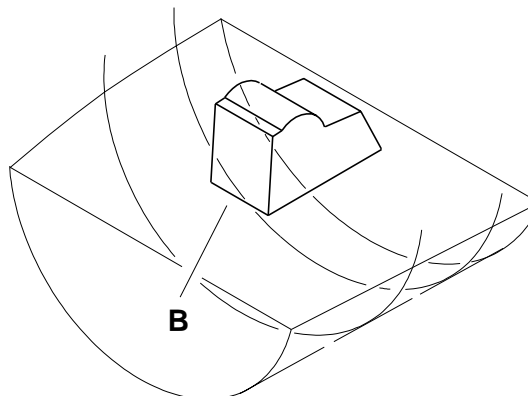
EFFECTIVITY: POST-MOD. S.B. 145-27-0075, AIRCRAFT WITH ELECTROMECHANICAL GUST LOCK SYSTEM

Forward Fuselage II - Cockpit Bottom - Electromechanical Gust Lock

Figure 606

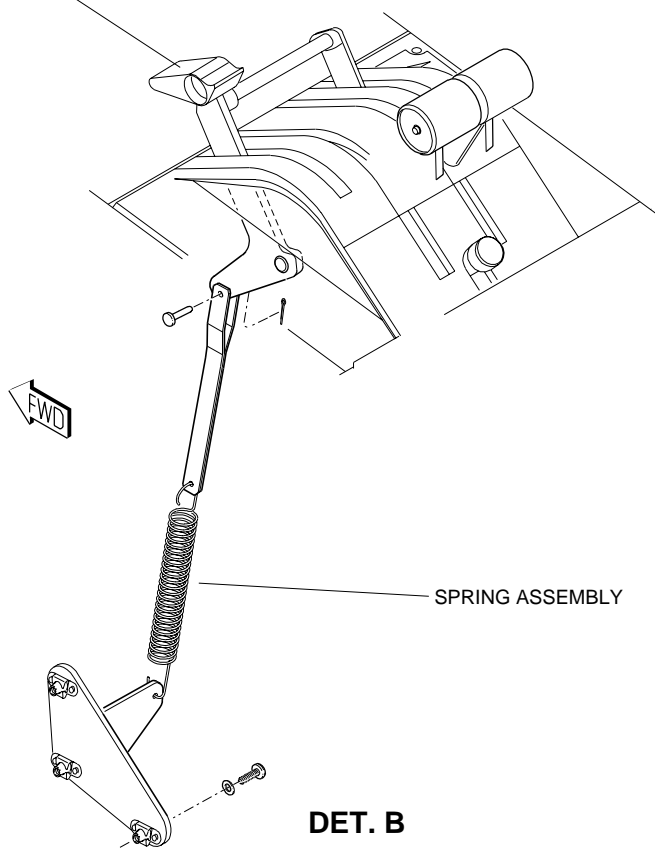


A
ZONE
123
124



DET. A

GUST LOCK LEVER
UNLOCKED



SPRING ASSEMBLY

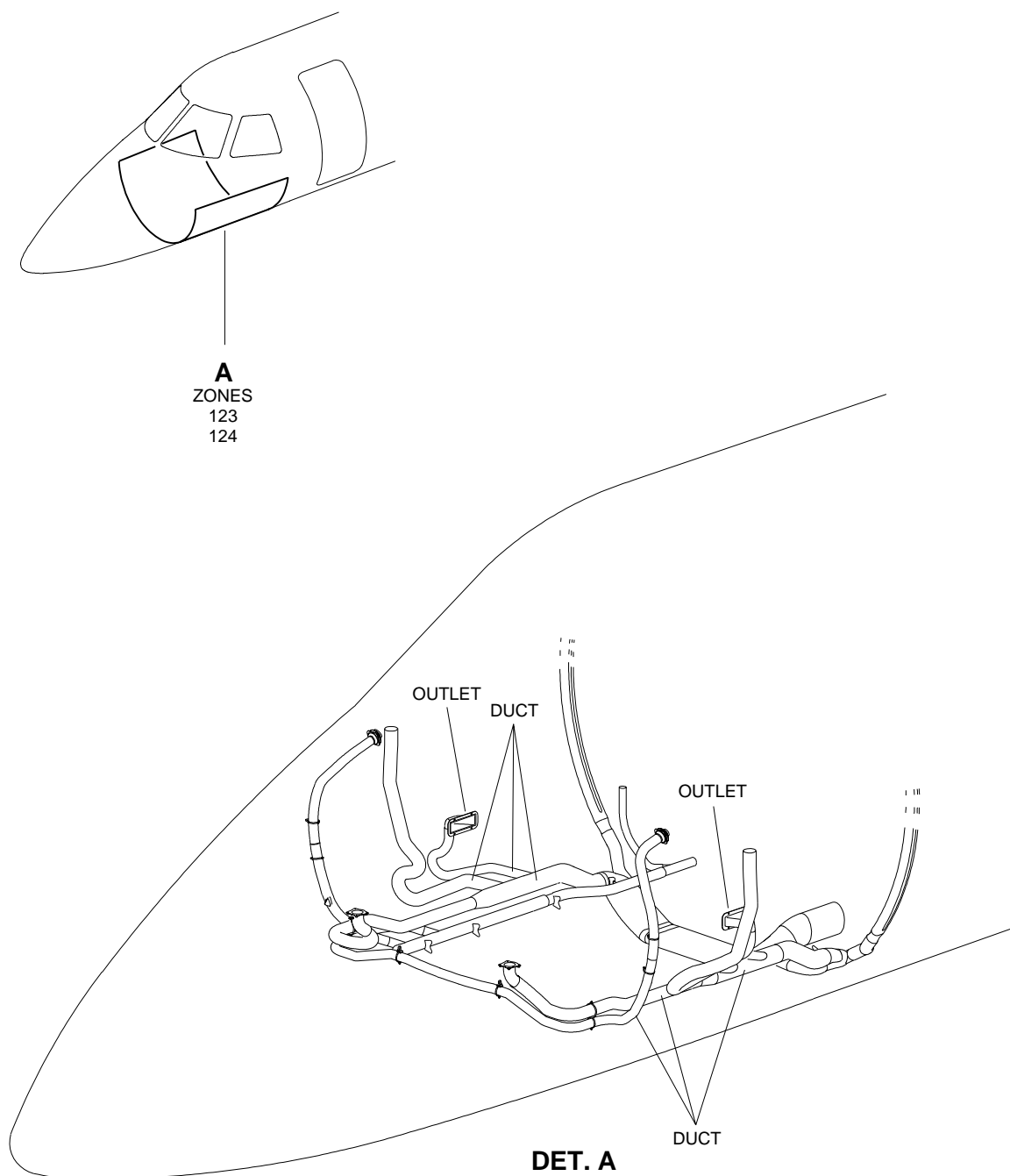
DET. B

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EFFECTIVITY: ALL

Forward Fuselage II - Cockpit Bottom - Air Conditioning

Figure 607



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