**INTEGRATED COMPUTERS - ADJUSTMENT/TEST**

*EFFECTIVITY: ALL*

1.   General

   A.   This section gives the procedures to do the test of the integrated computers.

   B.   The procedures in this section are given in the sequence below. The tasks identified with (♦) are part of the Scheduled Maintenance Requirements Document (SMRD).

| TASK NUMBER | DESCRIPTION | EFFECTIVITY |
|---|---|---|
| 31-42-00-700-801-A | INTEGRATED COMPUTERS - OPERA-TIONAL TEST | ALL |

TASK 31-42-00-700-801-A
*EFFECTIVITY:  ALL*

2.  INTEGRATED COMPUTERS - OPERATIONAL TEST

    A.  General

        (1)  This task gives the procedures to do the operational test of the integrated computers.

    B.  References

| REFERENCE | DESIGNATION |
|-----------|-------------|
| AMM SDS 23-12-00/1 | |
| AMM SDS 23-51-00/1 | |
| AMM SDS 23-81-00/1 | |
| AMM SDS 31-41-00/1 | |
| AMM SDS 31-42-00/1 | |
| AMM SDS 32-30-00/1 | |
| AMM SDS 34-22-00/1 | |
| AMM SDS 34-31-00/1 | |
| AMM SDS 34-32-00/1 | |
| AMM SDS 34-43-00/1 | |
| AMM SDS 34-51-00/1 | |
| AMM SDS 34-52-00/1 | |
| AMM SDS 34-53-00/1 | |
| AMM TASK 20-40-01-860-801-A/200 | ENERGIZATION OF THE AIRCRAFT WITH AN EXTERNAL POWER SOURCE |

    C.  Zones and Accesses

| ZONE | PANEL/DOOR | LOCATION |
|------|-----------|----------|
| 223/224 | | Cockpit |

    D.  Tools and Equipment

       Not Applicable

    E.  Auxiliary Items
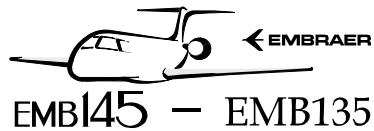
       Not Applicable

    F.  Consumable Materials

       Not Applicable

    G.  Expandable Parts

       Not Applicable

H.    Persons Recommended

| QTY | FUNCTION | PLACE |
|-----|----------|-------|
| 1 | Does the task | Cockpit |

I.    Preparation

*SUBTASK 841-002-A*

(1)    Make sure that the aircraft is safe for maintenance.

(2)    Energize the aircraft with the external DC power supply ( AMM TASK 20-40-01-860-801-A/200).

(3)    Make sure that these systems are operational and on:

–    VHF System (AMM SDS 23-12-00/1).

–    Airborne Audio System (AMM SDS 23-51-00/1).

–    Radio Management System (AMM SDS 23-81-00/1).

–    EICAS (AMM SDS 31-41-00/1).

–    Integrated Computer System (AMM SDS 31-42-00/1).

–    EFIS (AMM SDS 34-22-00/1).

–    Radio Altimeter System (AMM SDS 34-31-00/1).

–    VOR/ILS/GS/MB System (AMM SDS 34-32-00/1).

–    TCAS (AMM SDS 34-43-00/1).

–    DME System (AMM SDS 34-51-00/1).

–    Mode-S Transponder (AMM SDS 34-52-00/1).

–    ADF System (AMM SDS 34-53-00/1).

J.    Integrated Computers - Operational Test Procedures (Figure 501)  (Figure 502)  (Figure 503) (Figure 504)  (Figure 505)  (Figure 506)

*SUBTASK 710-002-A*

(1)    With the aircraft on the ground (WOW) (AMM SDS 32-30-00/1), do the test of the integrated computers as follows:

(a)    On the DC-550 1 display controller (left side) (AMM SDS 34-22-00/1), on the glareshield panel, use the RA knob (RA-TEST switch) to set 920 RA (DH) on PFD 1.

(b)    On the DC-550 1 display controller, on the glareshield panel, push and hold the RA-TEST switch.

Result:

<u>1</u>    The test indications of PFD 1 (first level), MFD 1, and EICAS come into view.

    2    After a few seconds, PFD 1 (left side) shows the Honeywell test page (second level), with the IC-600 IBIT indications and respective status.

    NOTE:  At this time, the status of the indications must change from FAIL to PASS. If the indication status stays in FAIL, you must repair the failure.

(c)    On DC-550 1, release the RA-TEST switch.

(d)    On the DC-550 2 display controller (right side) (AMM SDS 34-22-00/1), on the glareshield panel, use the RA knob (RA-TEST switch) to set 920 RA (DH) on PFD 2.

(e)    On the DC-550 2 display controller, push and hold the RA-TEST switch.
Result:

    1    The test indications of PFD 2 (first level) and MFD 2 come into view.

    2    After a few seconds, PFD 2 (right side) shows the Honeywell test page (second level), with the IC-600 IBIT indications and respective status.

    NOTE:  At this time, the status of the indications must change from FAIL to PASS. If the indication status stays in FAIL, you must repair the failure.
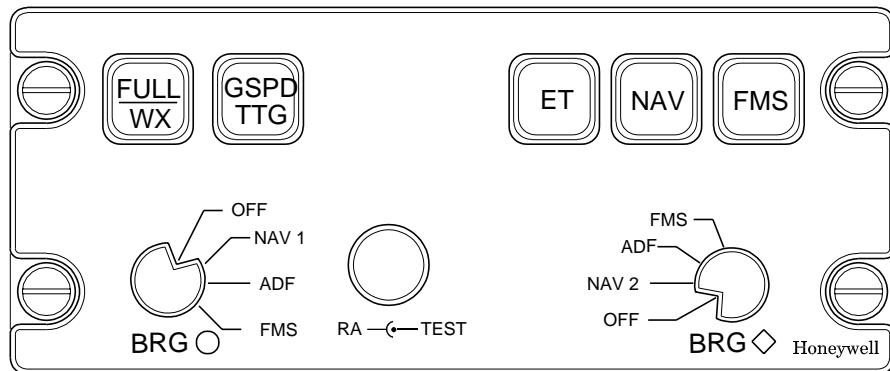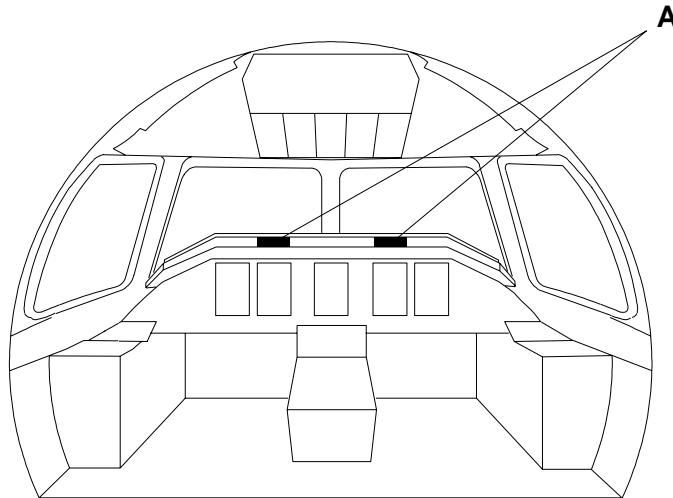
(f)    On DC-550 2, release the RA-TEST switch.

K.    Follow-on

*SUBTASK 842-002-A*

(1)    Deenergize the aircraft ( AMM TASK 20-40-01-860-801-A/200).

*EFFECTIVITY: ALL*
DC-550 Display Controller
Figure 501

A
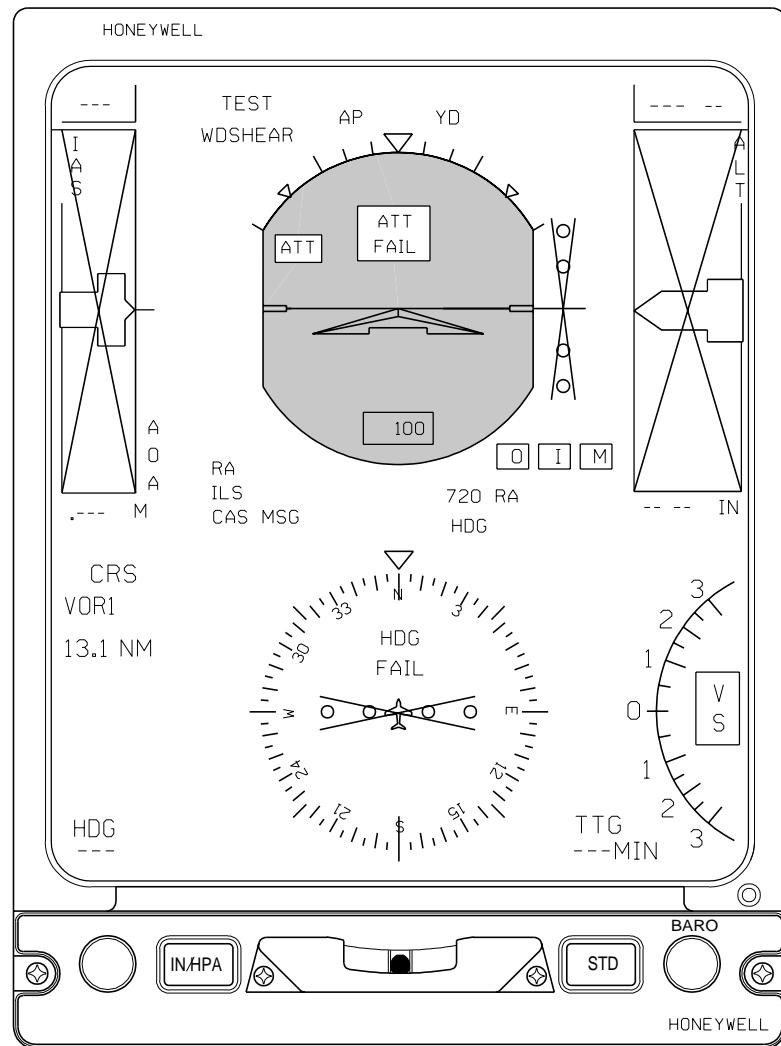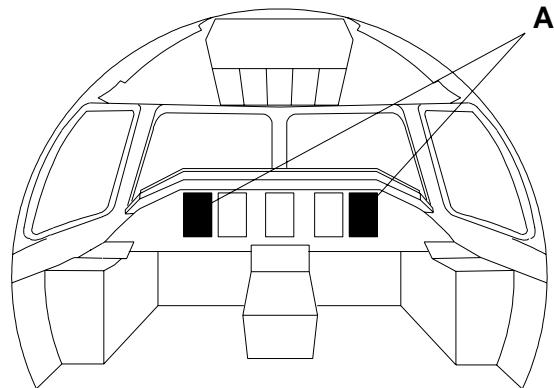
**DET. A**

145AMM340021.MCE A
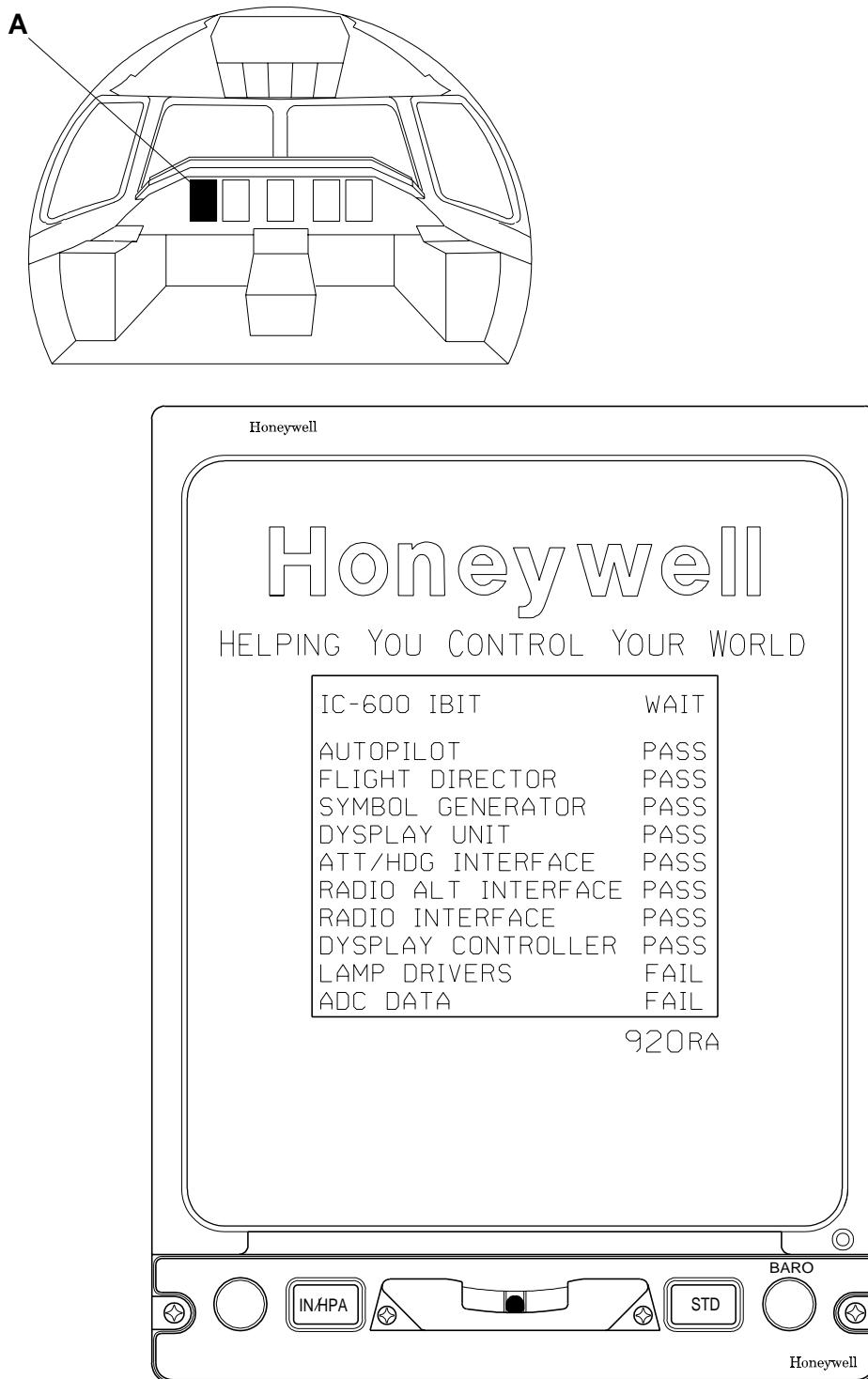
*EFFECTIVITY: ALL*
PFDs Test Indication - First Level
Figure 502



**DET. A**

145AMM310060.MCE A
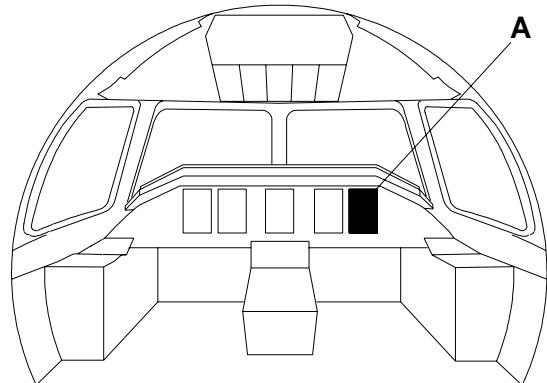
*EFFECTIVITY: ALL*
PFD 1 Test Indication - Second Level
Figure 503



```
              Honeywell


         Honeywell
  HELPING YOU CONTROL YOUR WORLD

    IC-600 IBIT              WAIT

    AUTOPILOT               PASS
    FLIGHT DIRECTOR         PASS
    SYMBOL GENERATOR        PASS
    DYSPLAY UNIT            PASS
    ATT/HDG INTERFACE       PASS
    RADIO ALT INTERFACE     PASS
    RADIO INTERFACE         PASS
    DYSPLAY CONTROLLER      PASS
    LAMP DRIVERS            FAIL
    ADC DATA                FAIL

                       920RA
```

DET. A

145AMM310061.MCE B

*EFFECTIVITY: ALL*
PFD 2 Test Indication - Second Level
Figure 504



```
Honeywell

Honeywell
HELPING  YOU  CONTROL  YOUR  WORLD

IC-600 IBIT              WAIT


FLIGHT  DIRECTOR        PASS
SYMBOL  GENERATOR       PASS
DYSPLAY  UNIT           PASS
ATT/HDG  INTERFACE      PASS
RADIO  ALT  INTERFACE   PASS
RADIO  INTERFACE        PASS
DYSPLAY  CONTROLLER     PASS
LAMP  DRIVERS           FAIL
ADC  DATA               FAIL

                   920RA
```

IN/HPA        STD    BARO

Honeywell
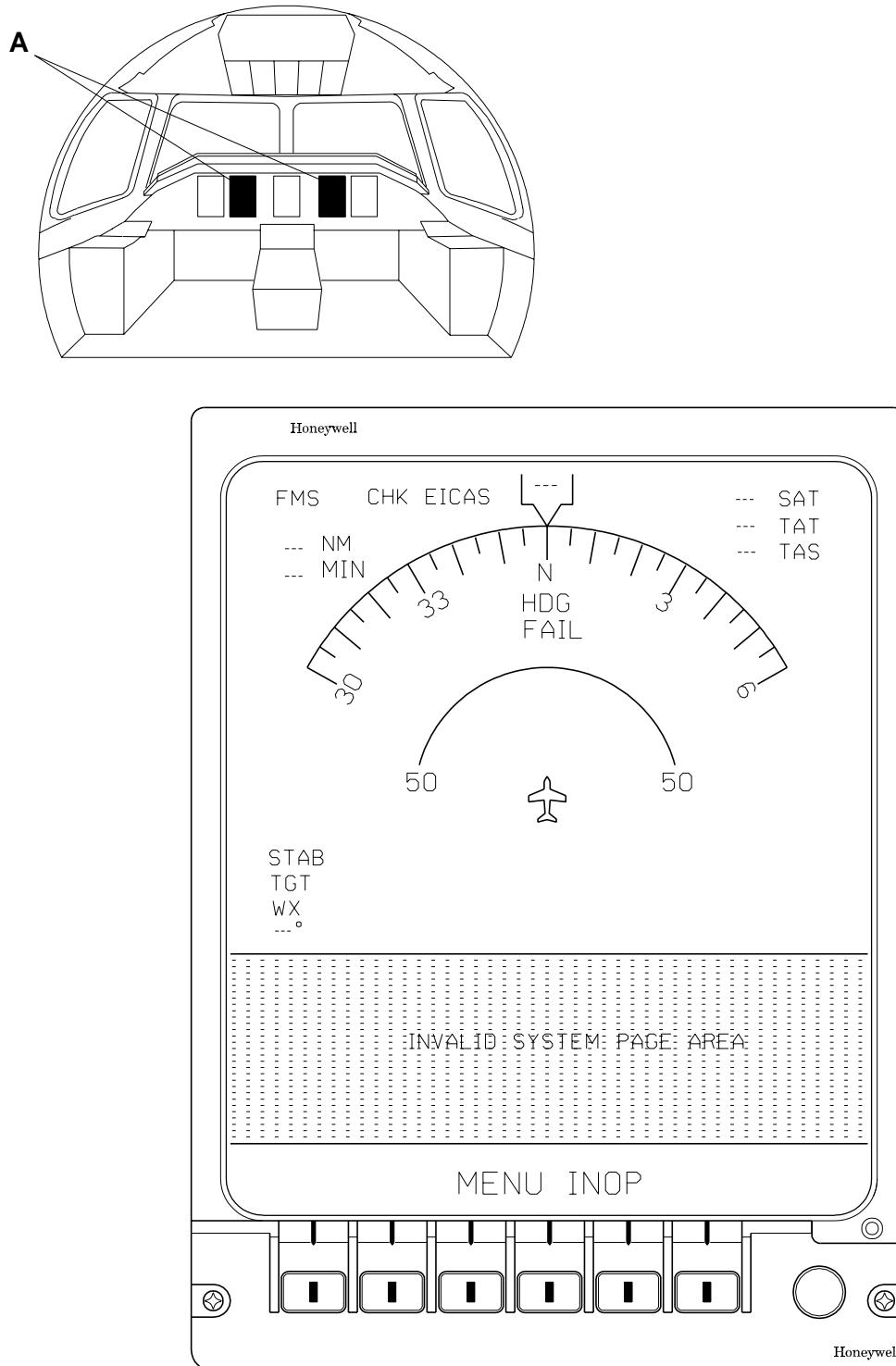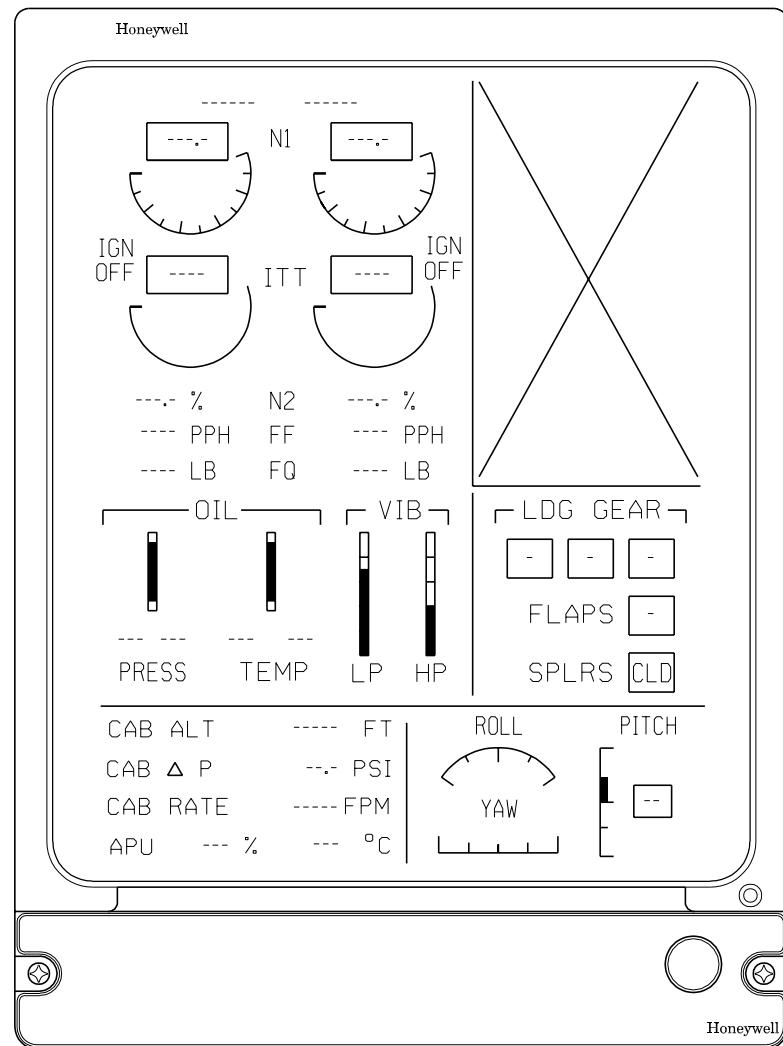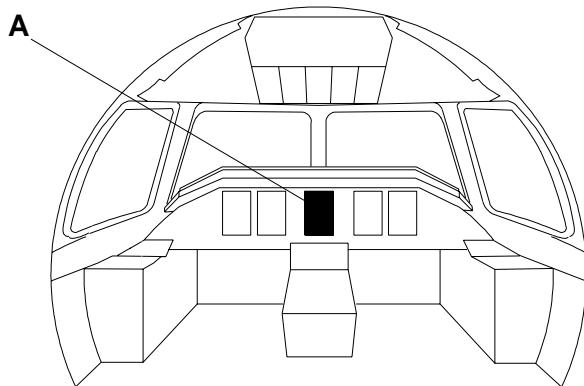
**DET. A**

145AMM310062.MCE A

*EFFECTIVITY: ALL*
MFDs Test Indication
Figure 505



A

Honeywell

FMS    CHK EICAS                    --- SAT
                                    --- TAT
--- NM                              --- TAS
--- MIN           N
           33    HDG        3
                 FAIL
       30                      6

         50              50


STAB
TGT
WX
---°


INVALID SYSTEM PAGE AREA


MENU INOP

Honeywell

**DET. A**

*EFFECTIVITY: ALL*
EICAS Test Indication
Figure 506



**DET. A**

145AMM310064.MCE A