



AIRCRAFT MAINTENANCE MANUAL

FUEL TEMPERATURE INDICATION - ADJUSTMENT/TEST

EFFECTIVITY: ALL

1. General

- A. This section gives the procedures to do the check of the fuel-temperature indication system.
- B. The procedures in this section are given in the sequence below. The tasks identified with (♦) are part of the Scheduled Maintenance Requirements Document (SMRD).

TASK NUMBER	DESCRIPTION	EFFECTIVITY
28-43-00-700-801-A ♦	FUEL-TEMPERATURE INDICATION SYSTEM - FUNCTIONAL CHECK	ACFT WITH WET WINGSTUB
28-43-00-700-802-A ♦	FUEL-TEMPERATURE INDICATION SYSTEM - FUNCTIONAL CHECK	ACFT WITH DRY WINGSTUB



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TASK 28-43-00-700-801-A

EFFECTIVITY: ACFT WITH WET WINGSTUB

2. FUEL-TEMPERATURE INDICATION SYSTEM - FUNCTIONAL CHECK

A. General

- (1) The access to the fuel temperature sensor is got through the LH landing gear bay.
- (2) A sensor housing prevents fuel loss when the temperature sensor is removed.

B. References

REFERENCE	DESIGNATION
AMM MPP 06-41-01/100	-
AMM TASK 20-40-01-860-801-A/200	ENERGIZATION OF THE AIRCRAFT WITH AN EXTERNAL POWER SOURCE
AMM TASK 28-41-00-200-801-A/600	-
AMM TASK 28-43-01-000-801-A/400	FUEL TEMPERATURE SENSOR - REMOVAL
AMM TASK 28-43-01-400-801-A/400	FUEL TEMPERATURE SENSOR - INSTALLATION
AMM TASK 32-00-01-910-801-A/200	LG SAFETY PIN - INSTALLATION AND REMOVAL

C. Zones and Accesses

ZONE	PANEL/DOOR	LOCATION
155	155DZ	Landing gear bay
155	155EZ	Landing gear bay

D. Tools and Equipment

Not Applicable

E. Auxiliary Items

ITEM	DESCRIPTION	PURPOSE	QTY
Commercially available	Standard thermometer	To measure the ambient temperature	1

F. Consumable Materials

Not Applicable

G. Expandable Parts

Not Applicable

H. Persons Recommended

QTY	FUNCTION	PLACE
1	Does the task	Landing gear bay



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I. Preparation

SUBTASK 841-002-A

- (1) Remove access panels 155DZ/155EZ (AMM MPP 06-41-01/100) to get access to the fuel sensor temperature.
- (2) Energize the aircraft with an External DC Power Supply ([AMM TASK 20-40-01-860-801-A/200](#)).

J. Functionally Check Fuel-Temperature Indication System ([Figure 501](#)) ([Figure 502](#))

SUBTASK 720-002-A

WARNING: MAKE SURE THAT THE LANDING GEAR SAFETY PINS ARE INSTALLED TO PREVENT INJURY TO PERSONS AND DAMAGE TO MATERIAL ([AMM TASK 32-00-01-910-801-A/200](#)).

- (1) Remove the fuel temperature sensor ([AMM TASK 28-43-01-000-801-A/400](#)) from its housing.
- (2) Hang the thermometer in the landing gear bay, near the sensor. After 10 minutes, read and write the value found.
 - (a) Compare the temperature found in step 2 with the value shown on the MFD display at Fuel page.
Result:
1 The maximum permitted difference between these values is $0^{\circ}\text{C} \pm 4^{\circ}\text{C}$.
- (3) Install the fuel temperature sensor ([AMM TASK 28-43-01-400-801-A/400](#)) in its housing.

K. Follow-on

SUBTASK 842-002-A

- (1) Deenergize the aircraft ([AMM TASK 20-40-01-860-801-A/200](#)).
- (2) Remove the thermometer in the landing gear bay.
- (3) Do an inspection on the fuel quantity indication harness (AMM TASK 28-41-00-200-801-A/600).

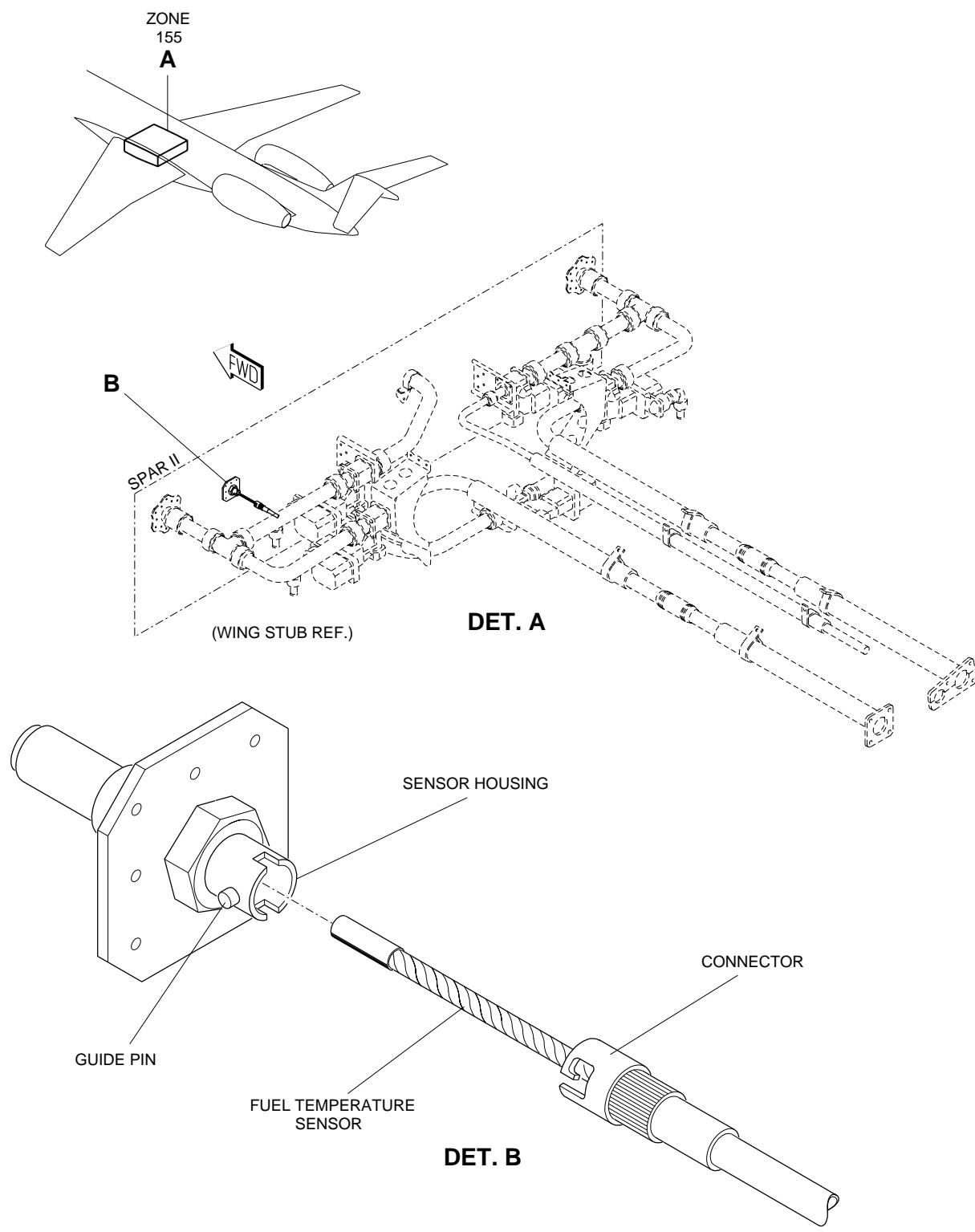
NOTE: The inspection of fuel quantity indication harness is a part of Critical Design Configuration Control Limitations (CDCCL) in the Airworthiness Limitations of the Maintenance Review Board Report (MRB).

- (4) Install access panels 155DZ/155EZ (AMM MPP 06-41-01/100).

EFFECTIVITY: ACFT WITH WET WINGSTUB

Fuel-Temperature Indication System - Component Locations

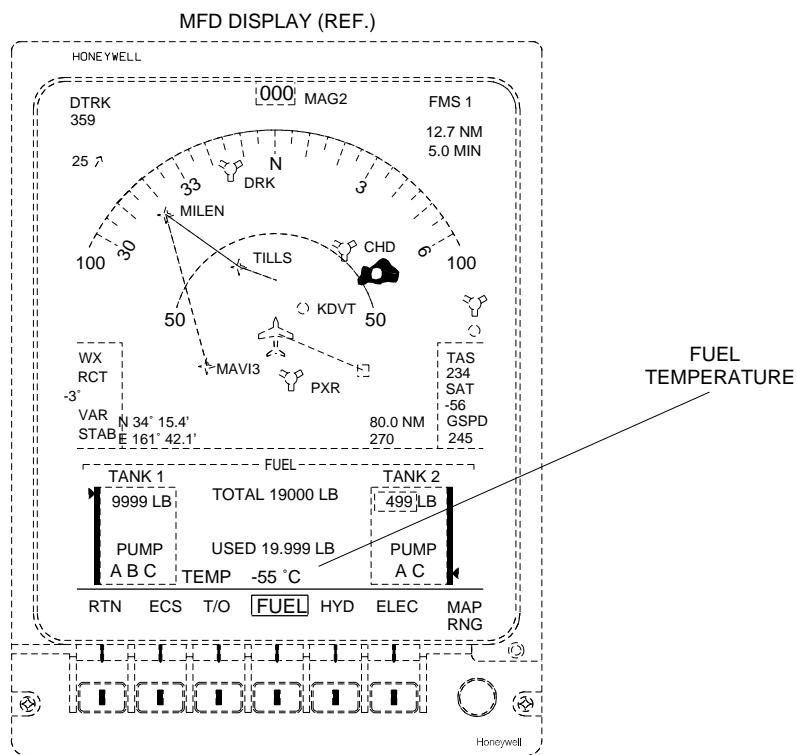
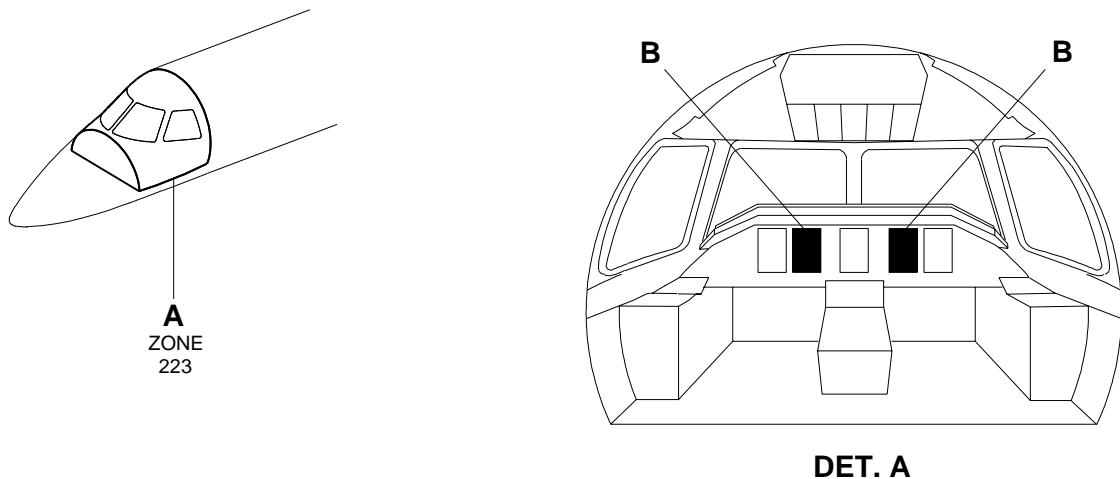
Figure 501



EFFECTIVITY: ACFT WITH WET WINGSTUB

Fuel-Temperature Indication System - Component Locations

Figure 502



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TASK 28-43-00-700-802-A
EFFECTIVITY: ACFT WITH DRY WINGSTUB
3. FUEL-TEMPERATURE INDICATION SYSTEM - FUNCTIONAL CHECK
A. General

- (1) The access to the fuel temperature sensor is got through the LH main landing-gear wheelwell.
- (2) A sensor housing prevents fuel loss when the temperature sensor is removed.

B. References

<i>REFERENCE</i>	<i>DESIGNATION</i>
AMM TASK 20-40-01-860-801-A/200	ENERGIZATION OF THE AIRCRAFT WITH AN EXTERNAL POWER SOURCE
AMM TASK 28-43-01-000-802-A/400	FUEL TEMPERATURE SENSOR - REMOVAL
AMM TASK 28-43-01-400-802-A/400	FUEL TEMPERATURE SENSOR - INSTALLATION
AMM TASK 32-00-01-910-801-A/200	LG SAFETY PIN - INSTALLATION AND REMOVAL

C. Zones and Accesses

Not Applicable

D. Tools and Equipment

Not Applicable

E. Auxiliary Items

<i>ITEM</i>	<i>DESCRIPTION</i>	<i>PURPOSE</i>	<i>QTY</i>
Commercially available	Standard thermometer	To measure the ambient temperature	1

F. Consumable Materials

Not Applicable

G. Expandable Parts

Not Applicable

H. Persons Recommended

<i>QTY</i>	<i>FUNCTION</i>	<i>PLACE</i>
1	Does the task	LH MLG Wheelwell/Cockpit

I. Preparation
SUBTASK 841-003-A

- (1) Energize the aircraft with an External DC Power Supply ([AMM TASK 20-40-01-860-801-A/200](#)).

J. Functionally Check Fuel-Temperature Indication System ([Figure 503](#)) ([Figure 502](#))

SUBTASK 720-003-A

WARNING: MAKE SURE THAT THE LANDING GEAR SAFETY PINS ARE INSTALLED TO PREVENT INJURY TO PERSONS AND DAMAGE TO MATERIAL (AMM TASK 32-00-01-910-801-A/200).

- (1) Remove the fuel temperature sensor ([AMM TASK 28-43-01-000-802-A/400](#)) from its housing.
- (2) Hang the thermometer in the LH MLG wheelwell, near the sensor. After 10 minutes, read and write the value found.
 - (a) Compare the temperature found in step 2 with the value shown on the MFD display at Fuel page.
Result:
1 The maximum permitted difference between these values is $0^{\circ}\text{C} \pm 4^{\circ}\text{C}$.
- (3) Install the fuel temperature sensor ([AMM TASK 28-43-01-400-802-A/400](#)) in its housing.

K. Follow-on

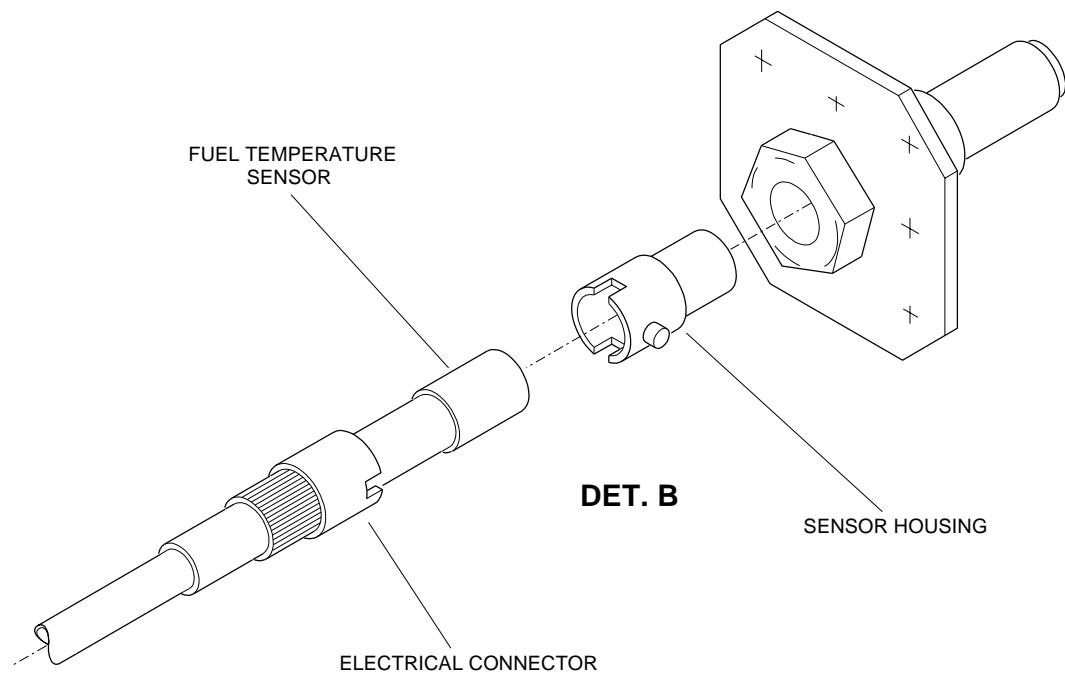
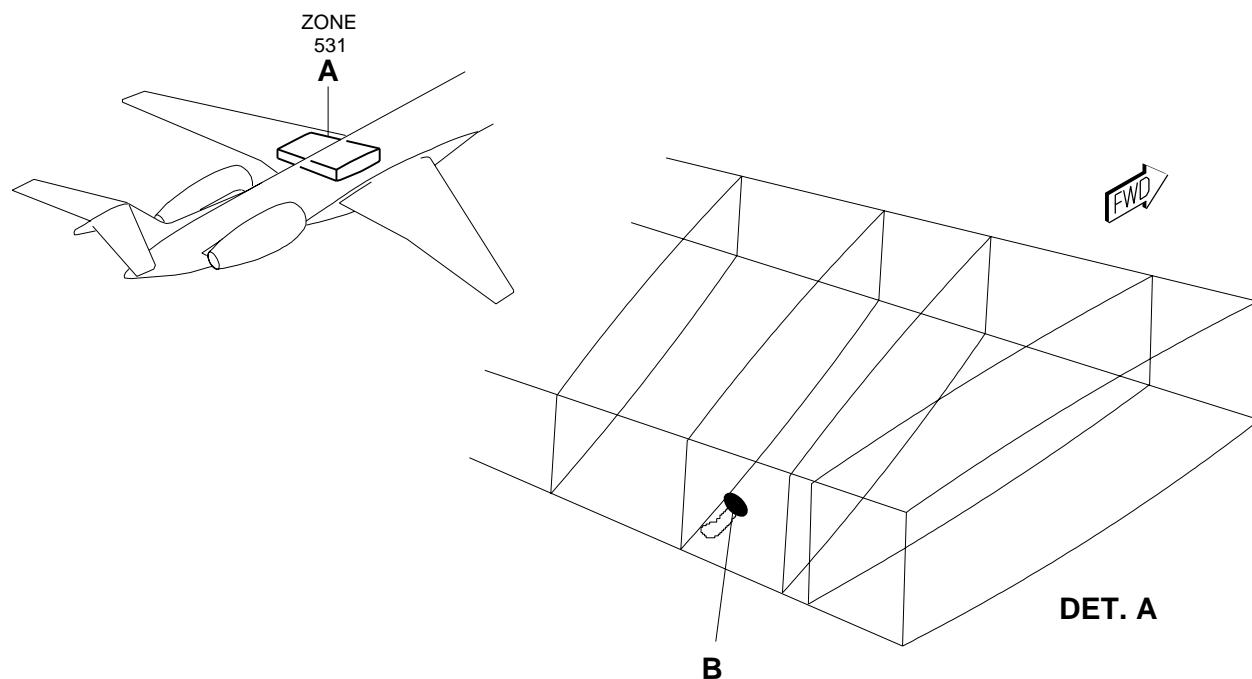
SUBTASK 842-003-A

- (1) Deenergize the aircraft ([AMM TASK 20-40-01-860-801-A/200](#)).
- (2) Remove the thermometer from the LH MLG wheelwell.

EFFECTIVITY: ACFT WITH DRY WINGSTUB

Fuel-Temperature Indication System - Component Locations

Figure 503



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