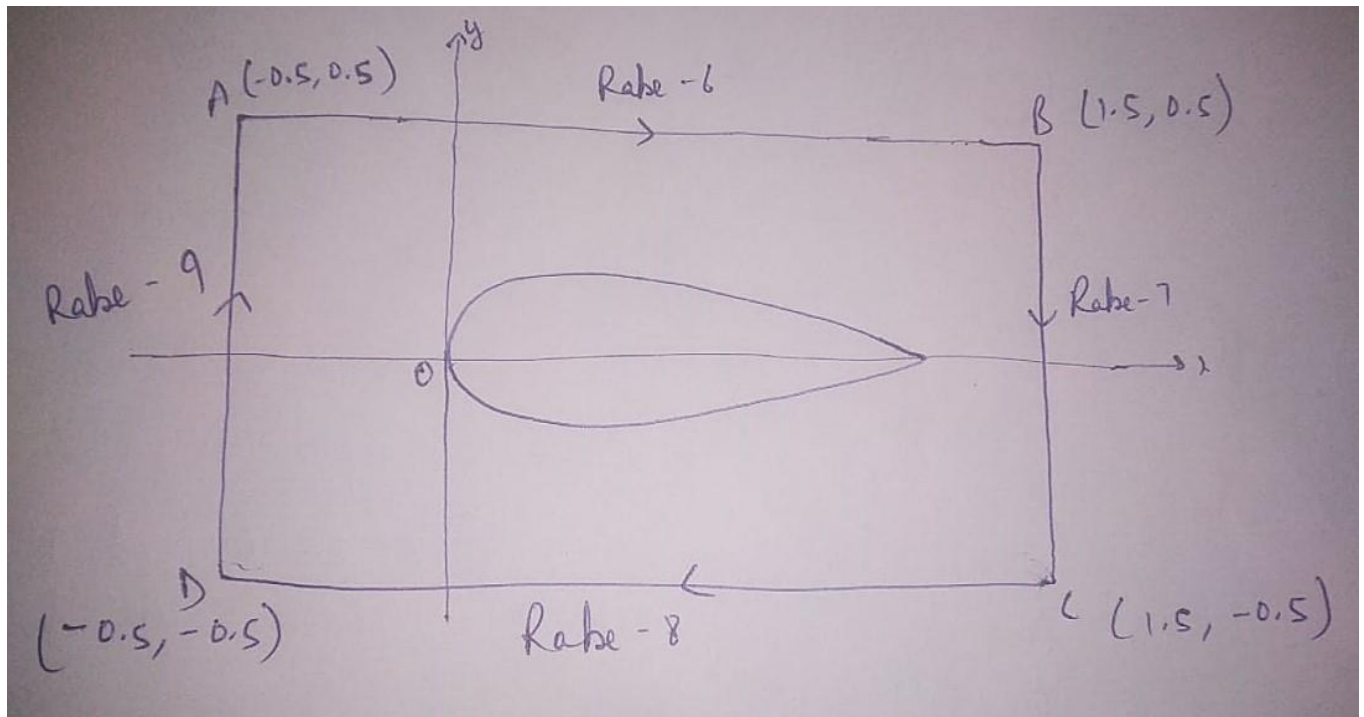


Name	Harsh Sharma
Roll no.	19AE30022

NACA 0012 symmetric airfoil

For 0° angle of attack



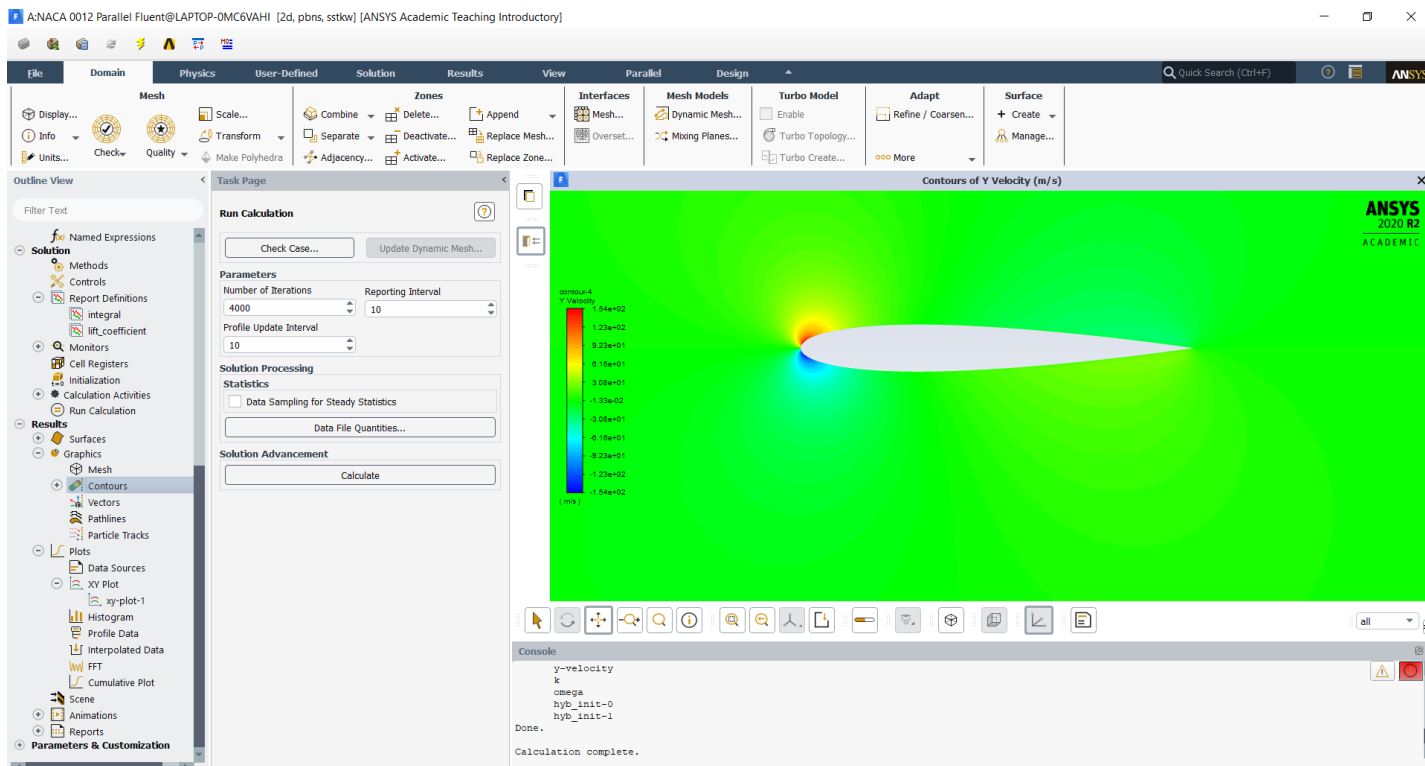
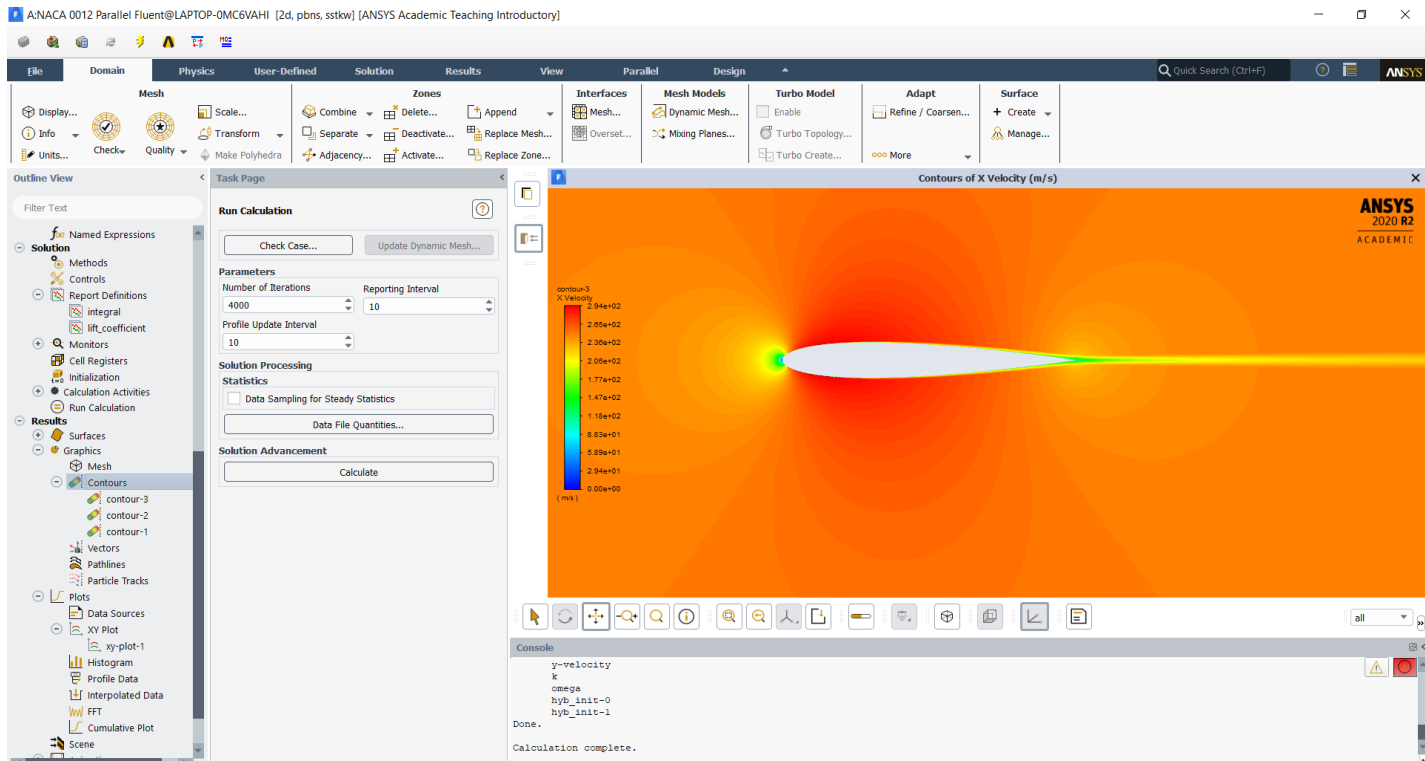
$$\begin{aligned}
 \text{Circulation} &= \int_A^B \mathbf{v} \cdot d\mathbf{l} + \int_B^C \mathbf{v} \cdot d\mathbf{l} + \int_C^D \mathbf{v} \cdot d\mathbf{l} + \int_D^A \mathbf{v} \cdot d\mathbf{l} \\
 &= 223.9 - 17.66 - 169.8 + 43.93 \\
 &= 80.37
 \end{aligned}$$

$$\text{Air density} = 1.225 \text{ Kg/m}^3$$

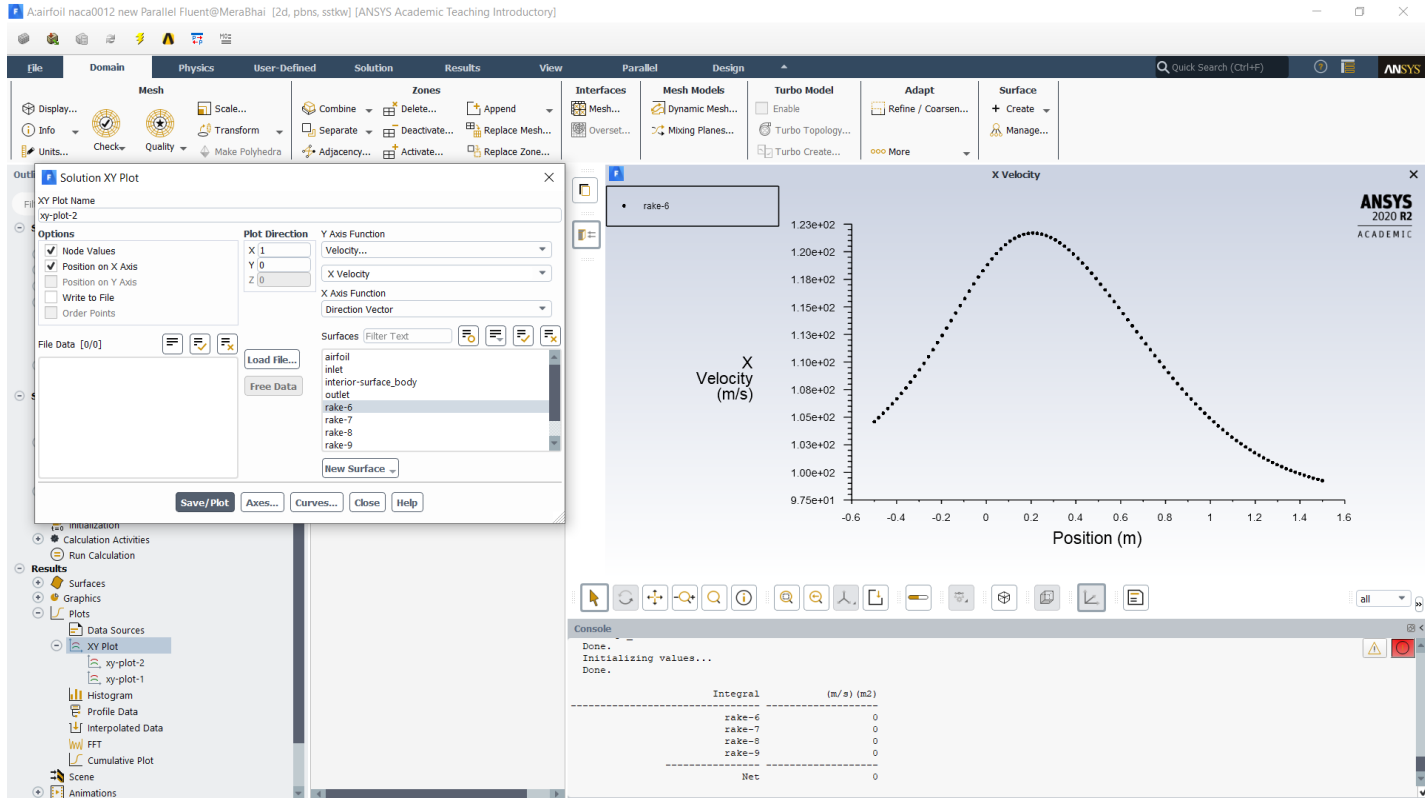
$$\text{Freestream velocity} = 250 \text{ m/s}$$

$$\text{Lift} = (\text{Air density}) * (\text{Freestream velocity}) * (\text{circulation})$$

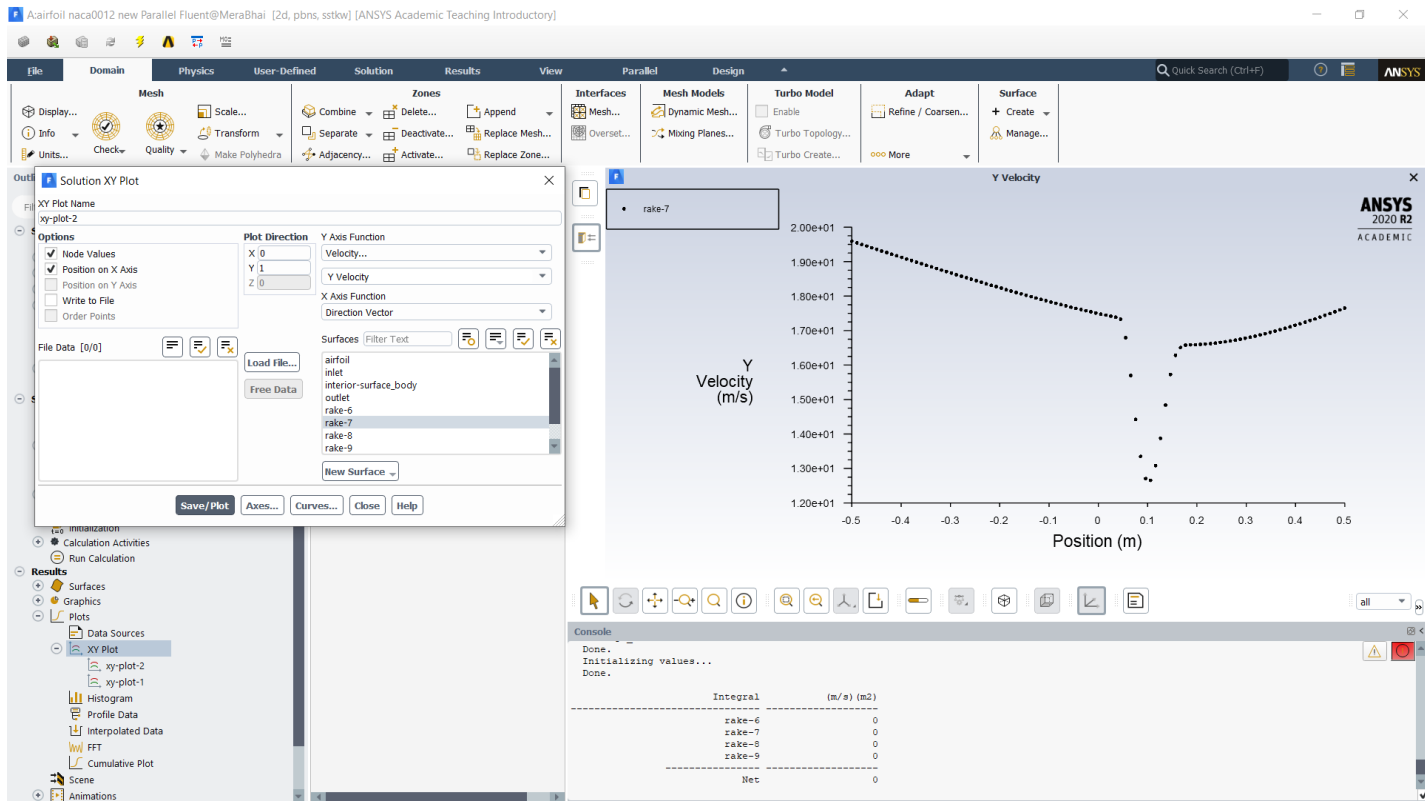
$$= 1.225 * 250 * 80.37 = 24613.3125 \text{ N}$$



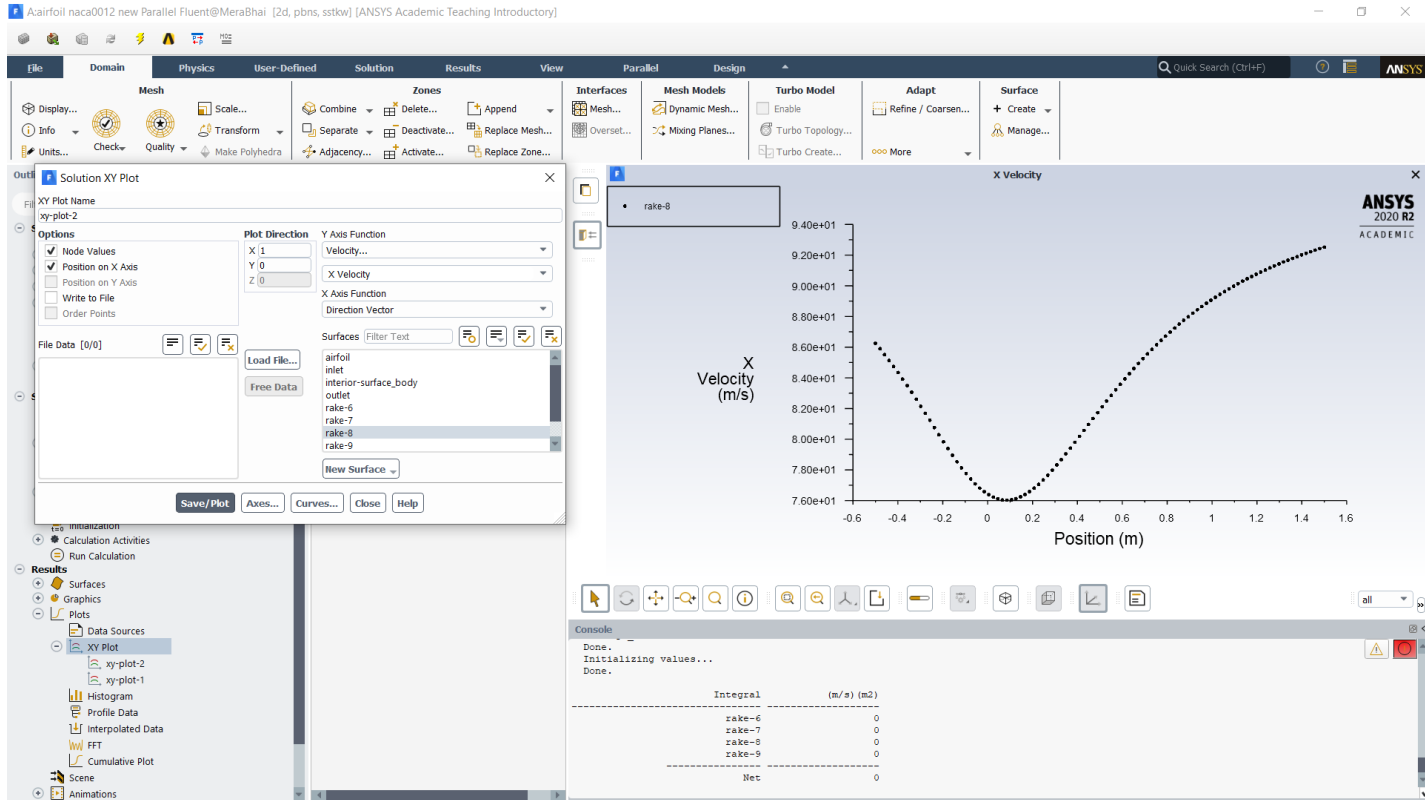
Rake-6



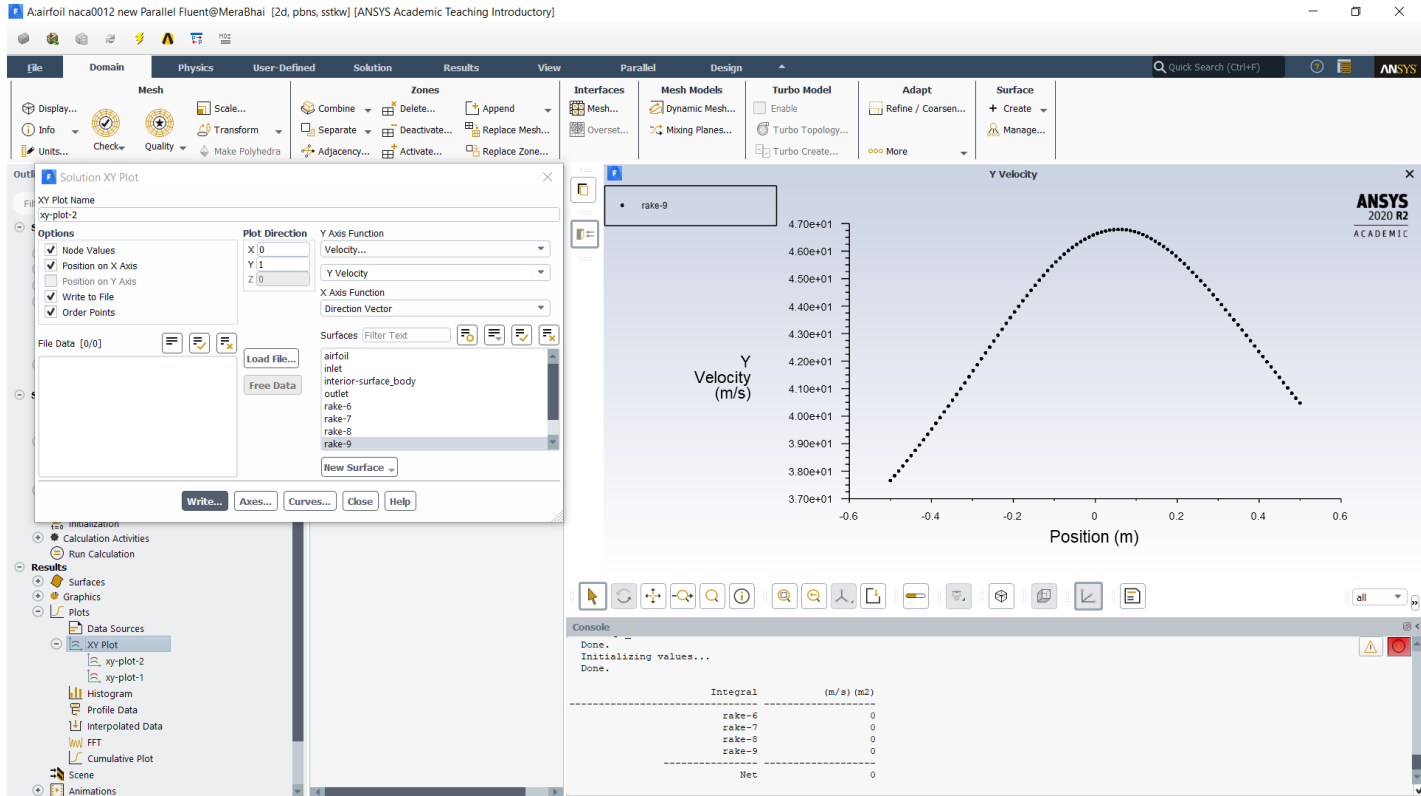
Rake-7



Rake-8



Rake-9



For 3° angle of attack

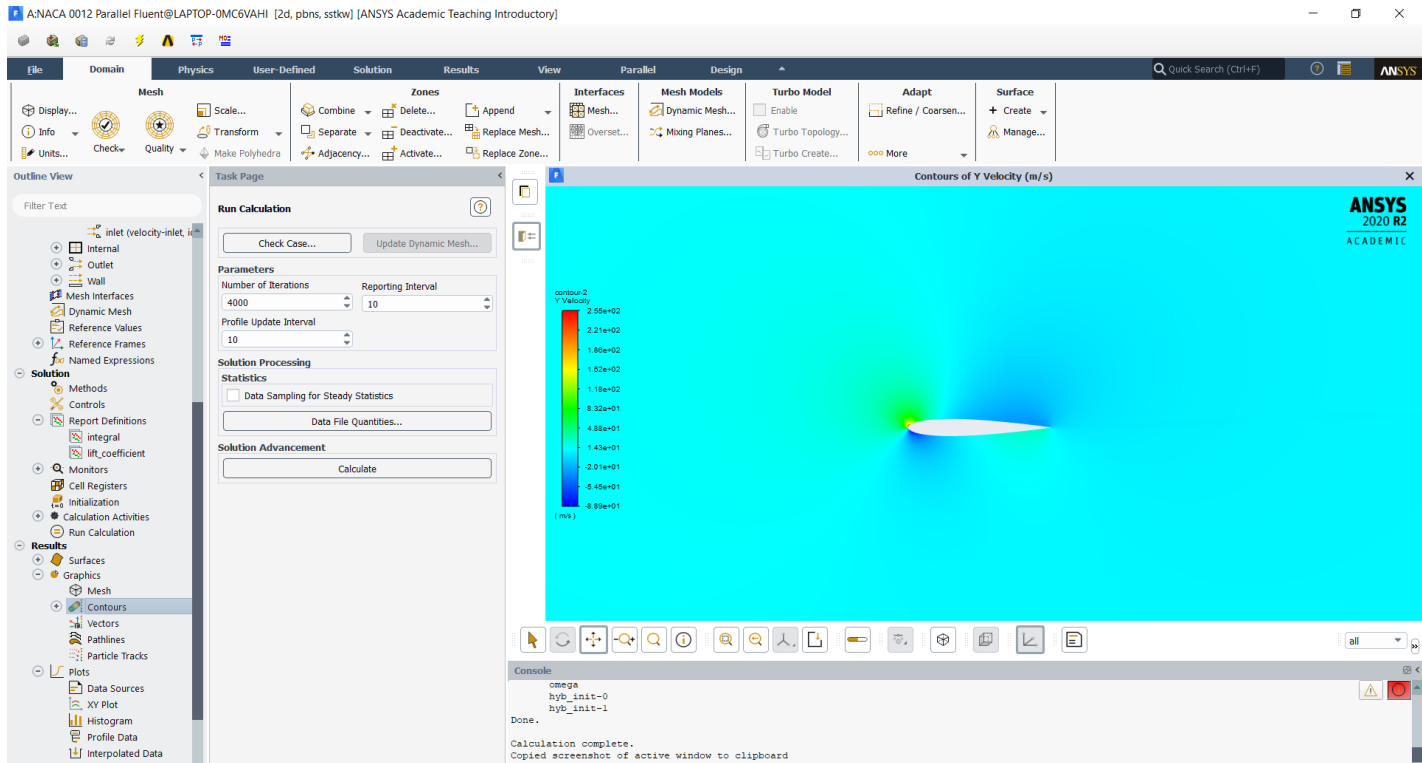
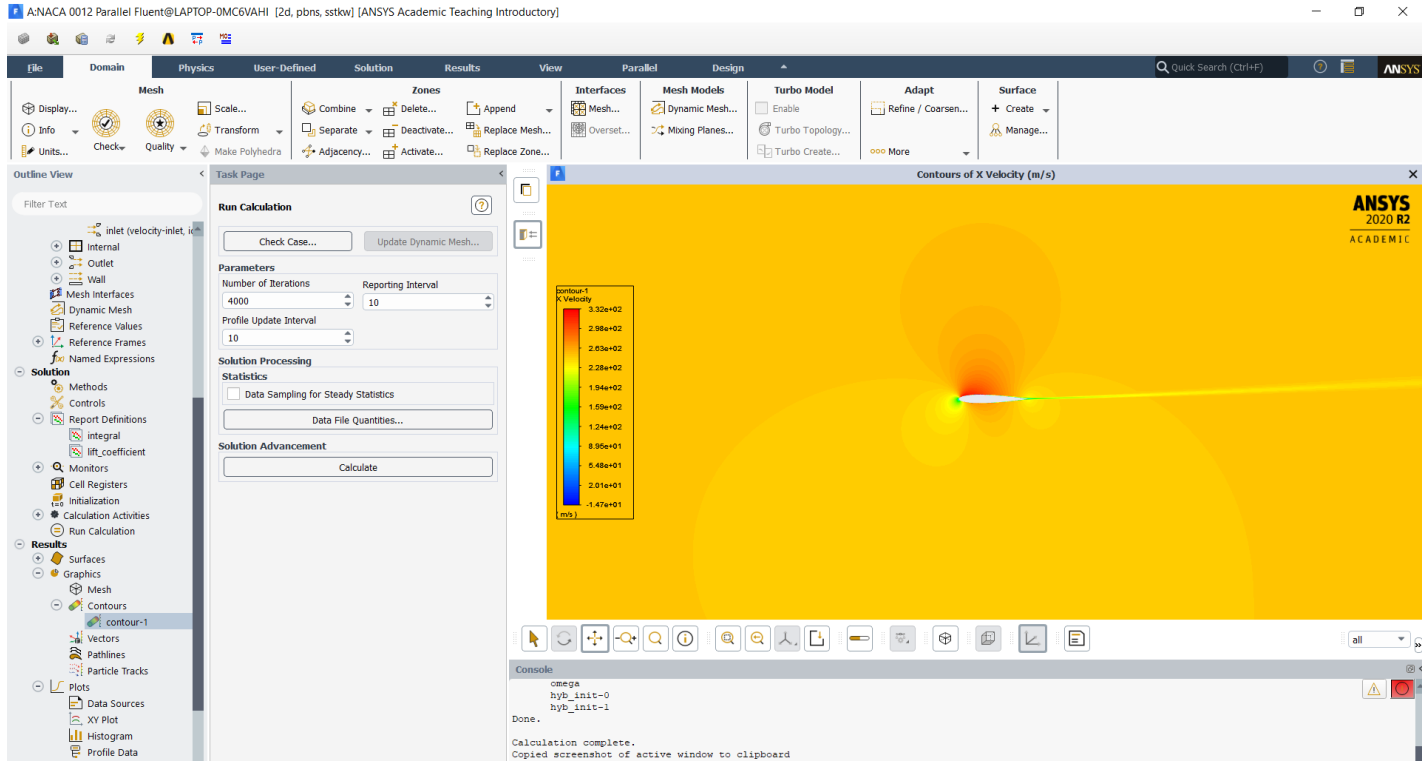
$$\begin{aligned} \text{Circulation} &= \int_A^B \mathbf{v} \cdot d\mathbf{l} + \int_B^C \mathbf{v} \cdot d\mathbf{l} + \int_C^D \mathbf{v} \cdot d\mathbf{l} + \int_D^A \mathbf{v} \cdot d\mathbf{l} \\ &= 524.435 - 7.94 - 496.595 + 21.37 \\ &= 41.27 \end{aligned}$$

Air density = 1.225 Kg/m³

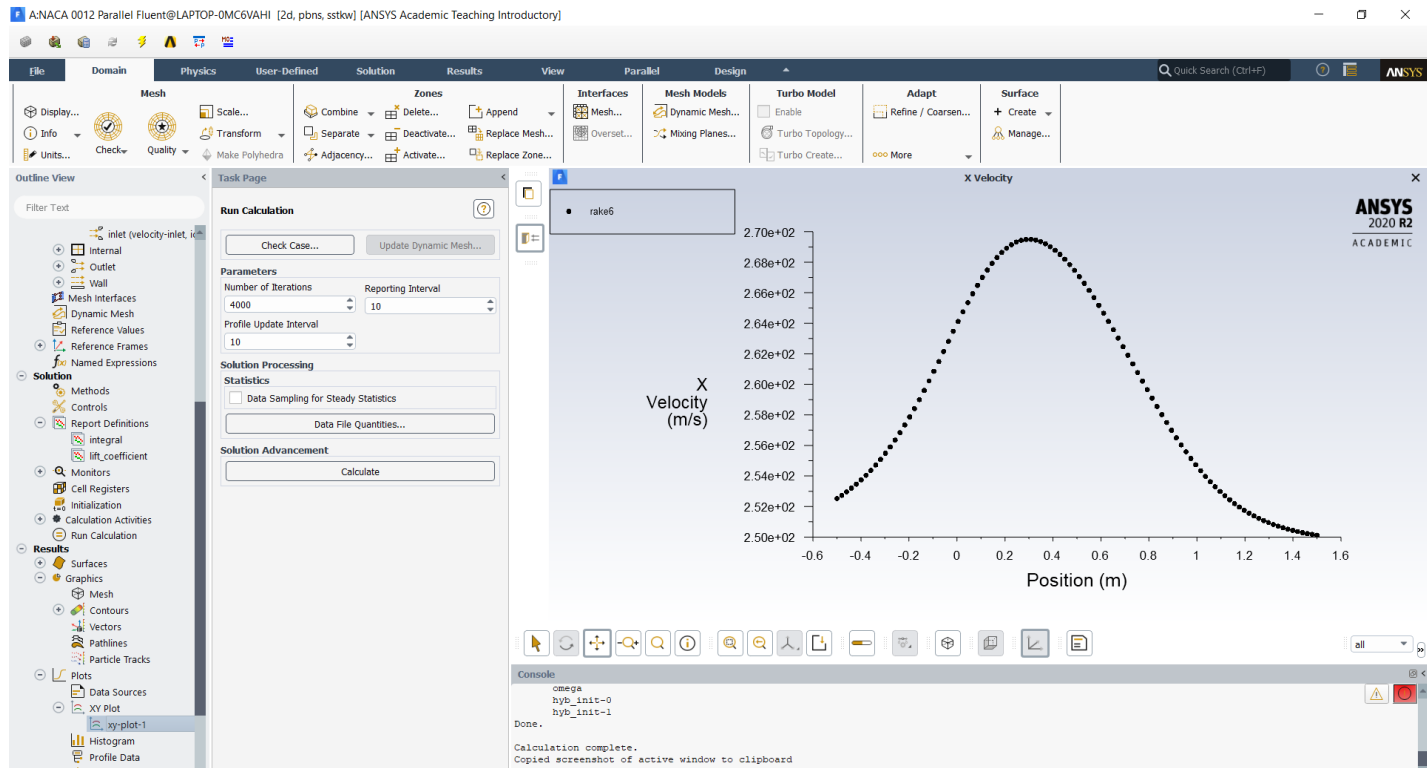
Freestream velocity = 250m/s

Lift = (Air density)*(Freestream velocity)*(circulation)

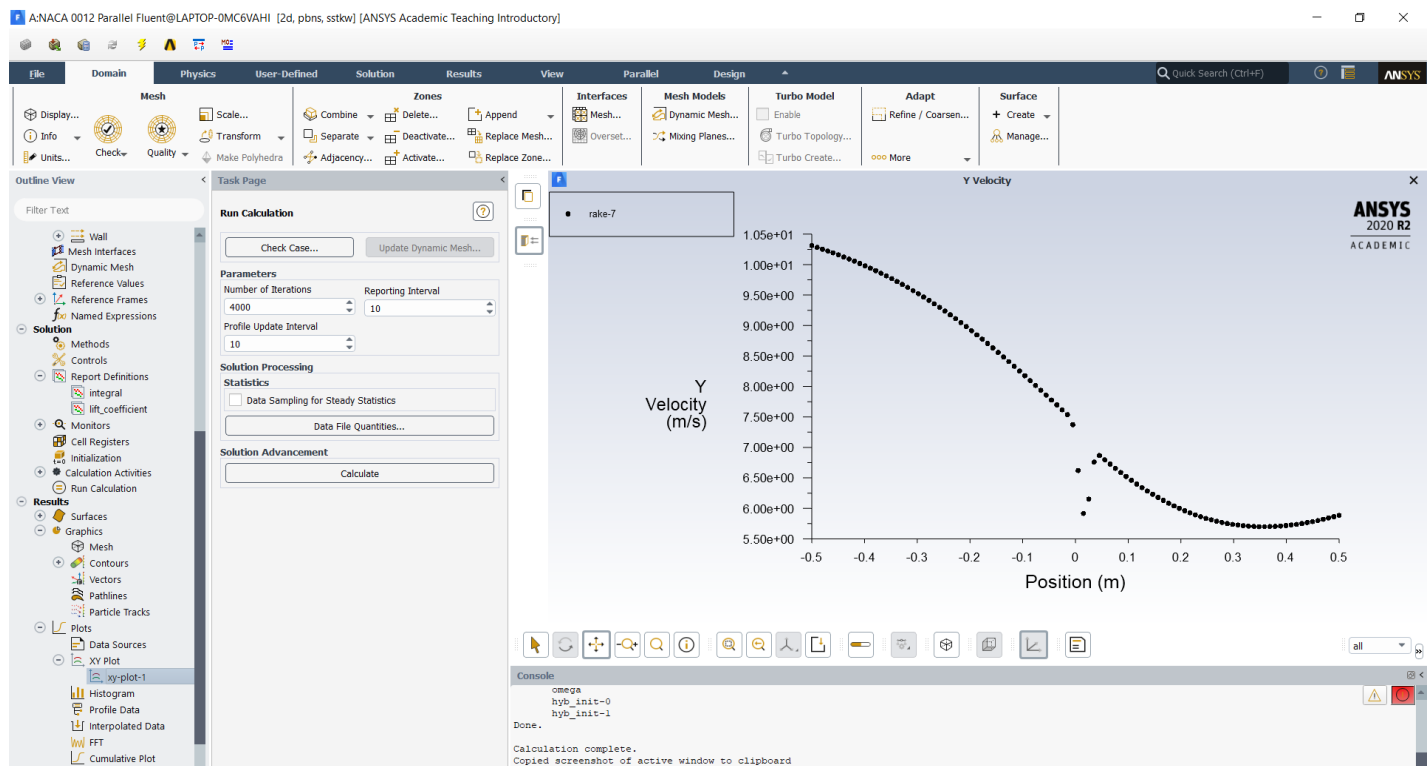
$$= 1.225 * 250 * 41.27 = 12638.94 \text{ N}$$



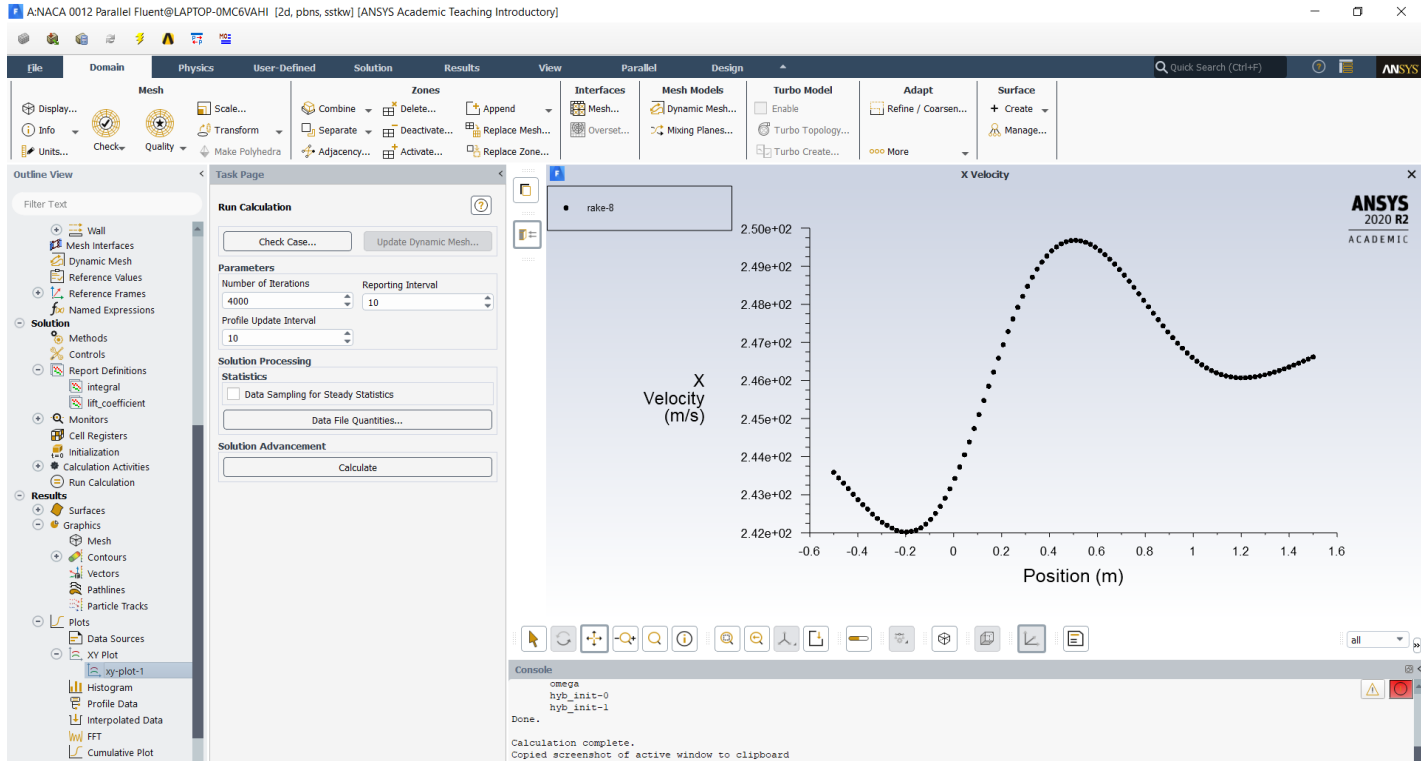
Rake-6



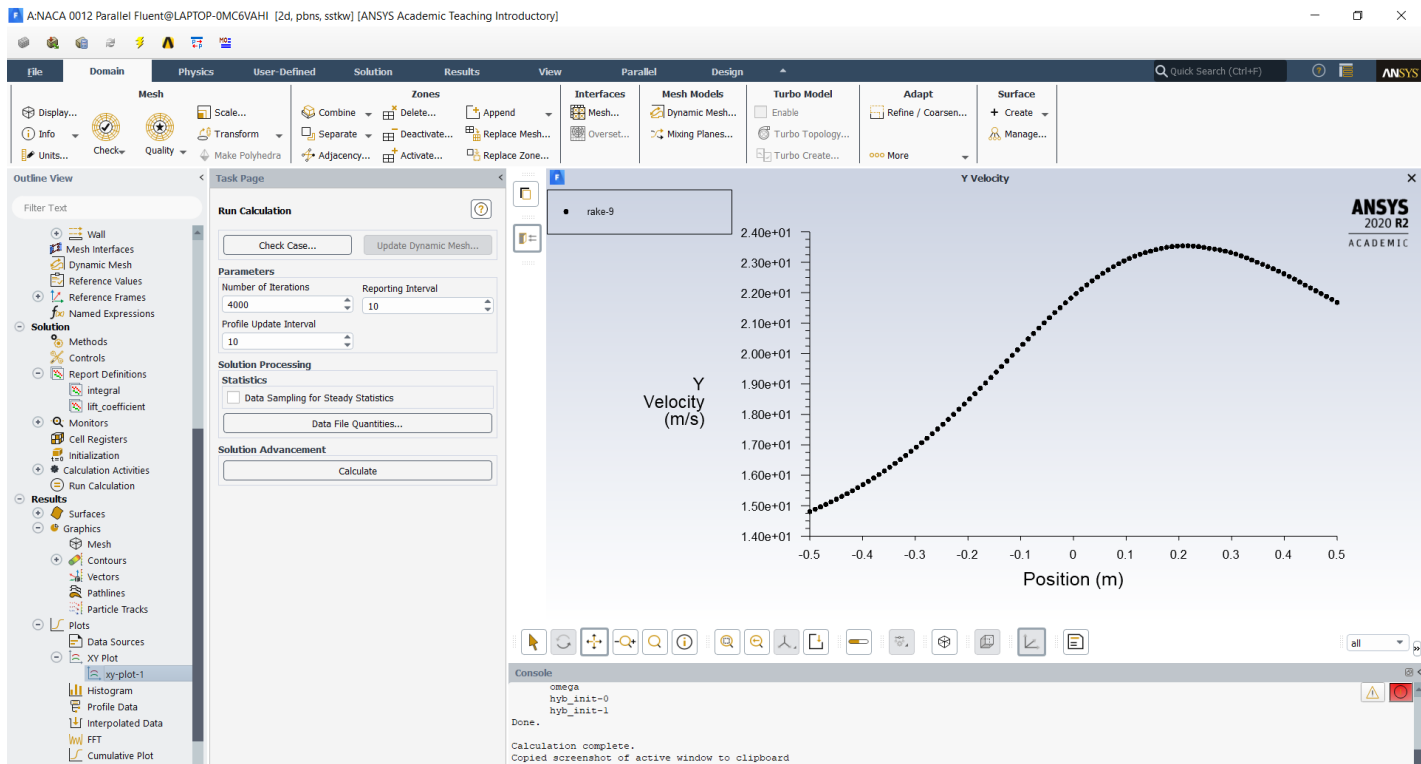
Rake-7



Rake-8



Rake-9



NACA 4412 Cambered Airfoil

For 0° angle of attack

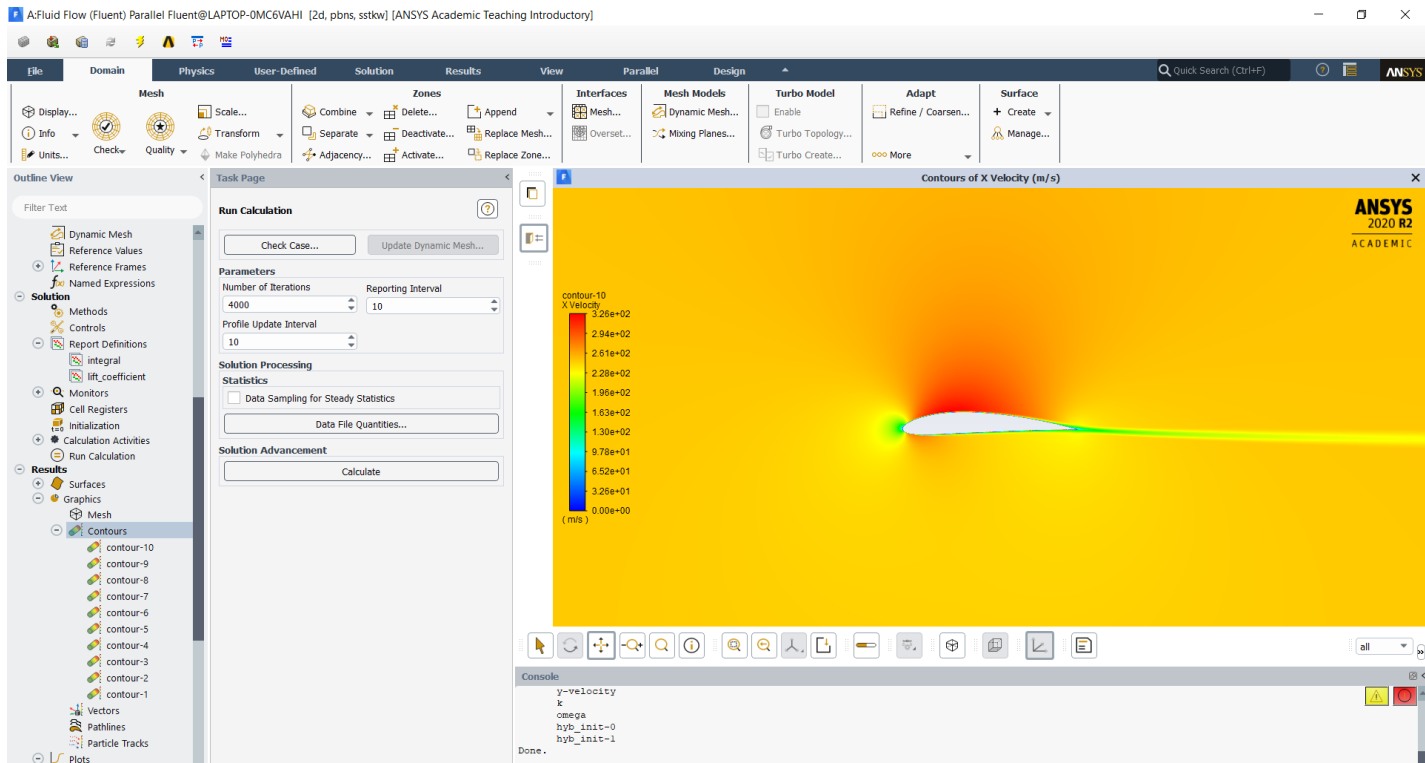
$$\begin{aligned}\text{Circulation} &= \int_A^B \mathbf{v} \cdot d\mathbf{l} + \int_B^C \mathbf{v} \cdot d\mathbf{l} + \int_C^D \mathbf{v} \cdot d\mathbf{l} + \int_D^A \mathbf{v} \cdot d\mathbf{l} \\ &= 530.51 - (-8.18) - 493.07 + 8 \\ &= 53.62\end{aligned}$$

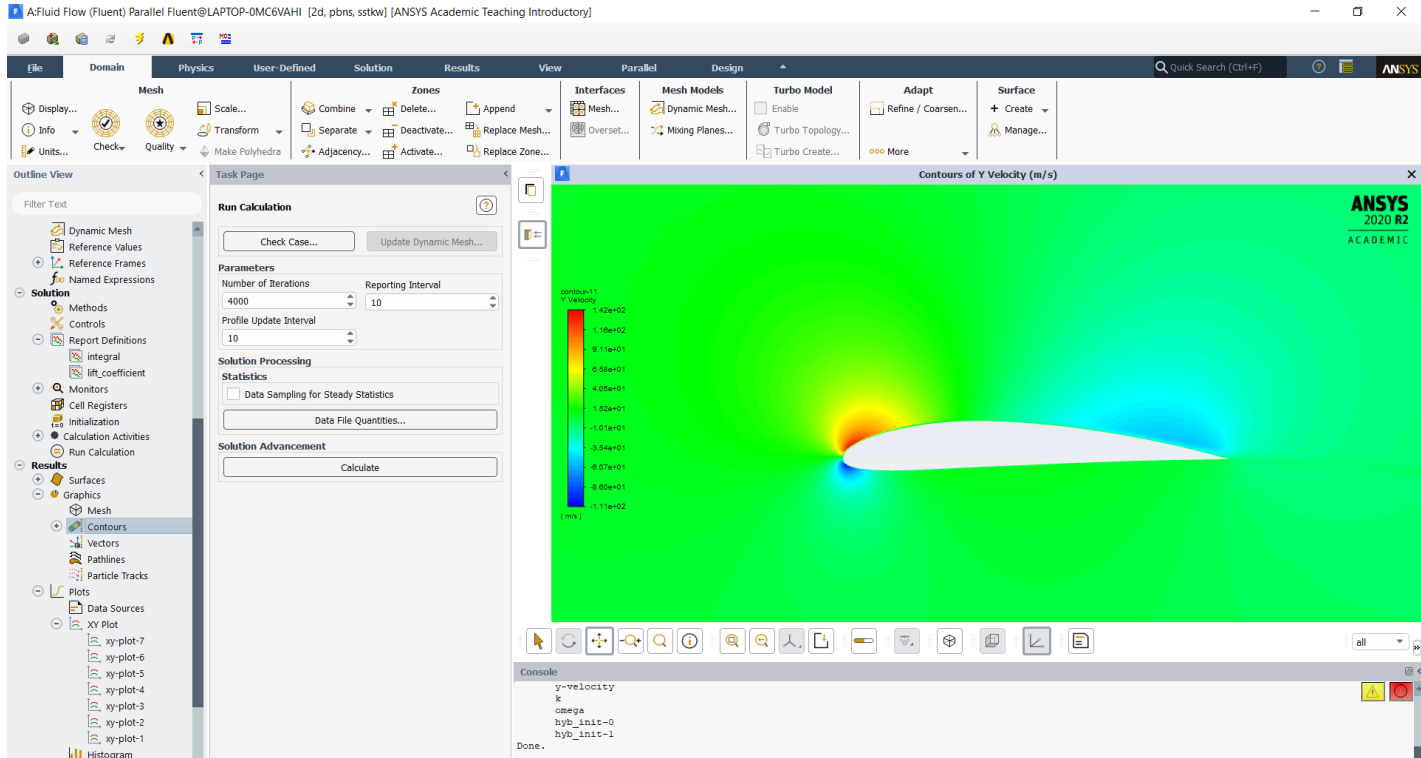
Air density = 1.225 Kg/m³

Freestream velocity = 250m/s

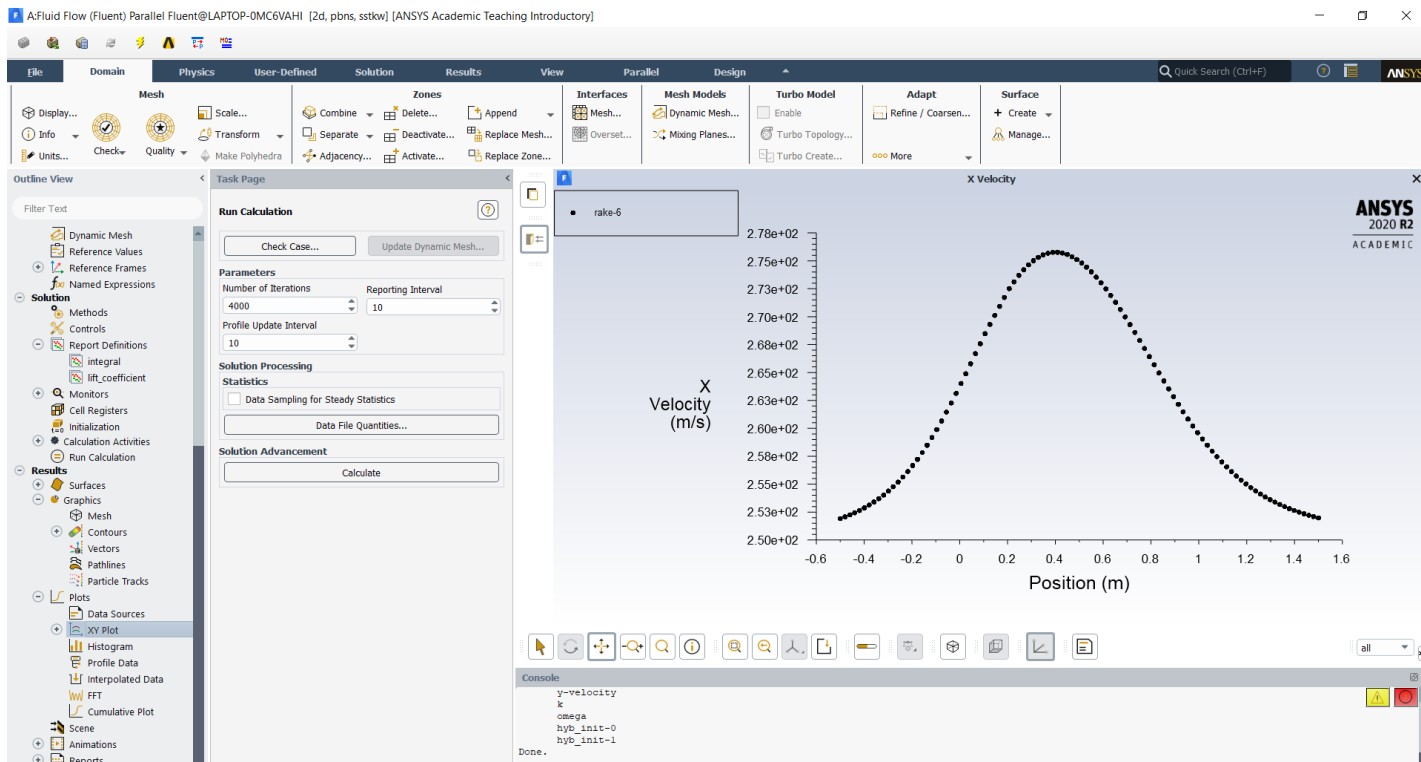
Lift = (Air density)*(Freestream velocity)*(circulation)

$$= 1.225 * 250 * 53.62 = 16421.125 \text{ N}$$

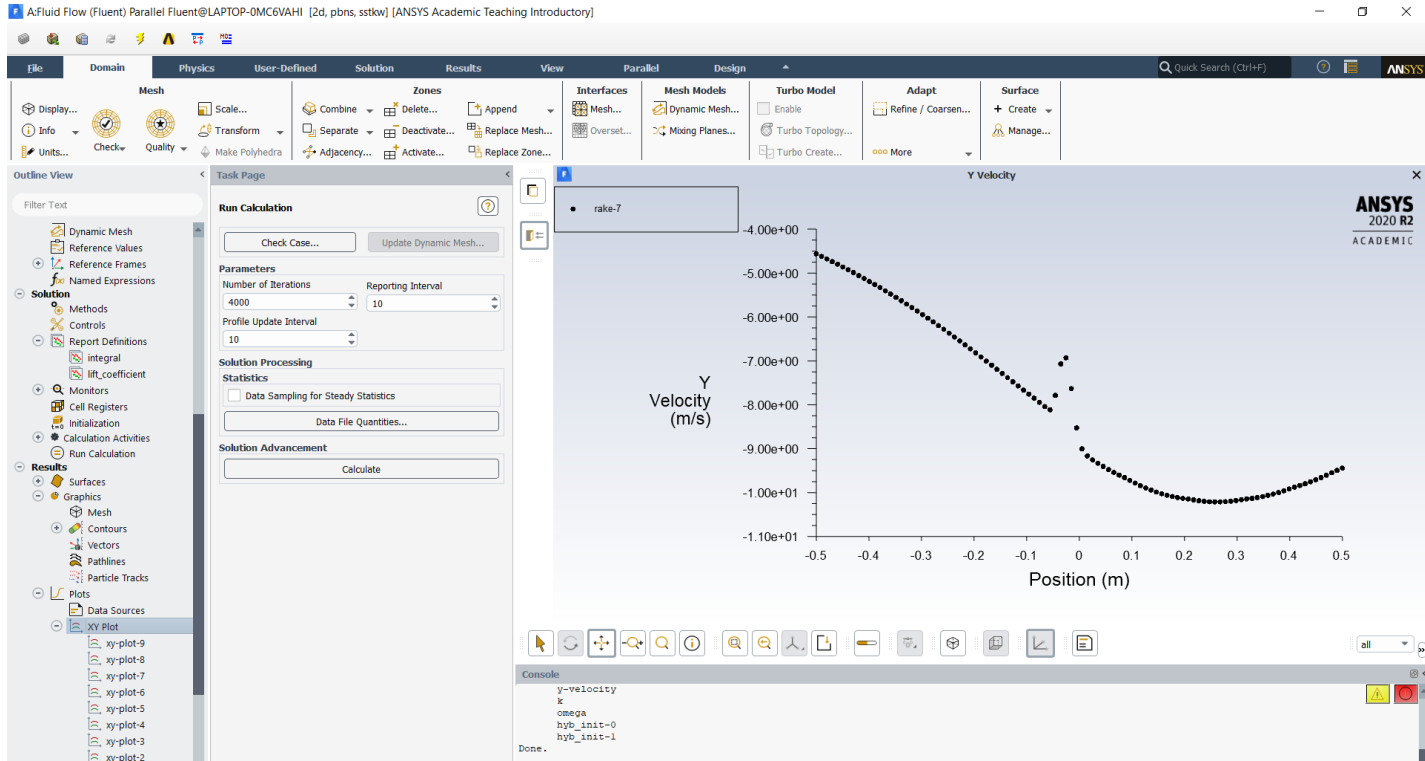




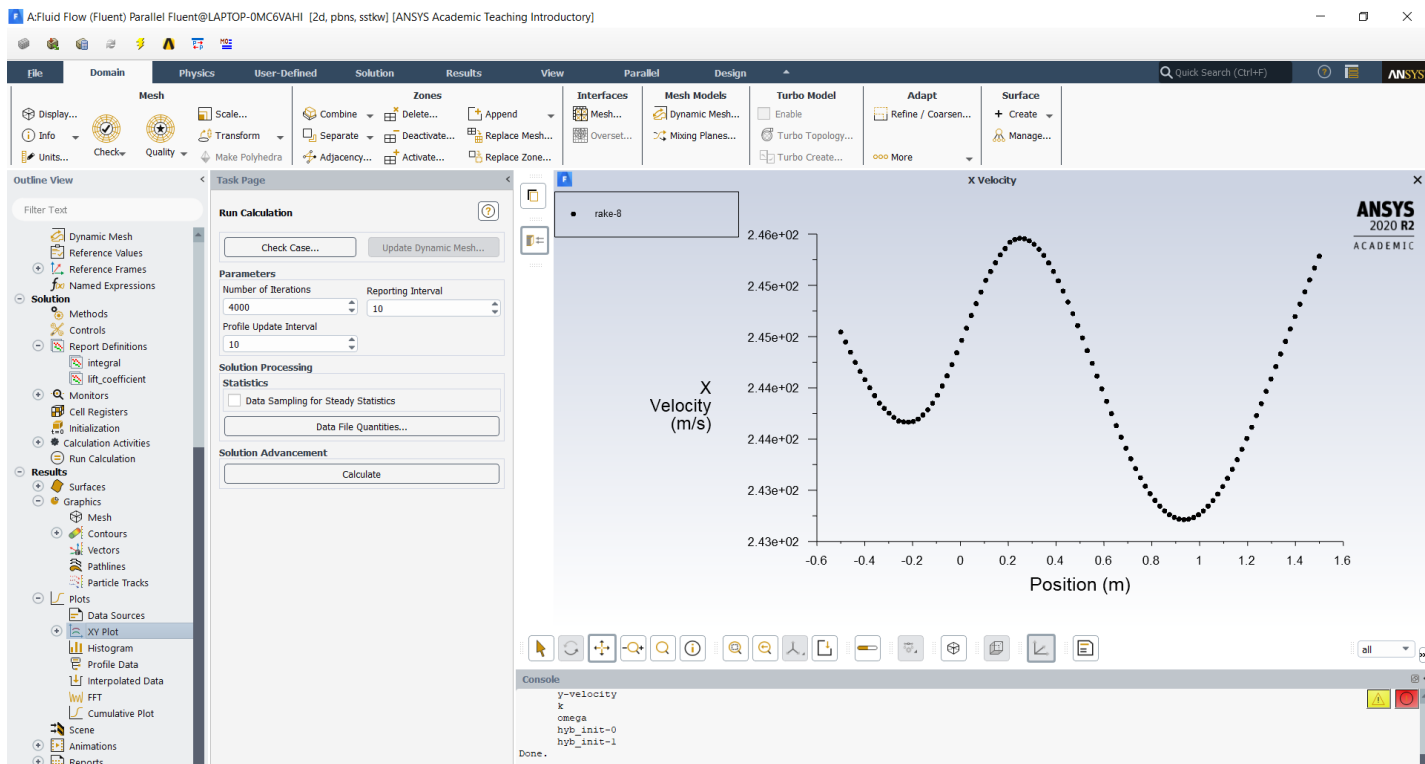
Rake-6



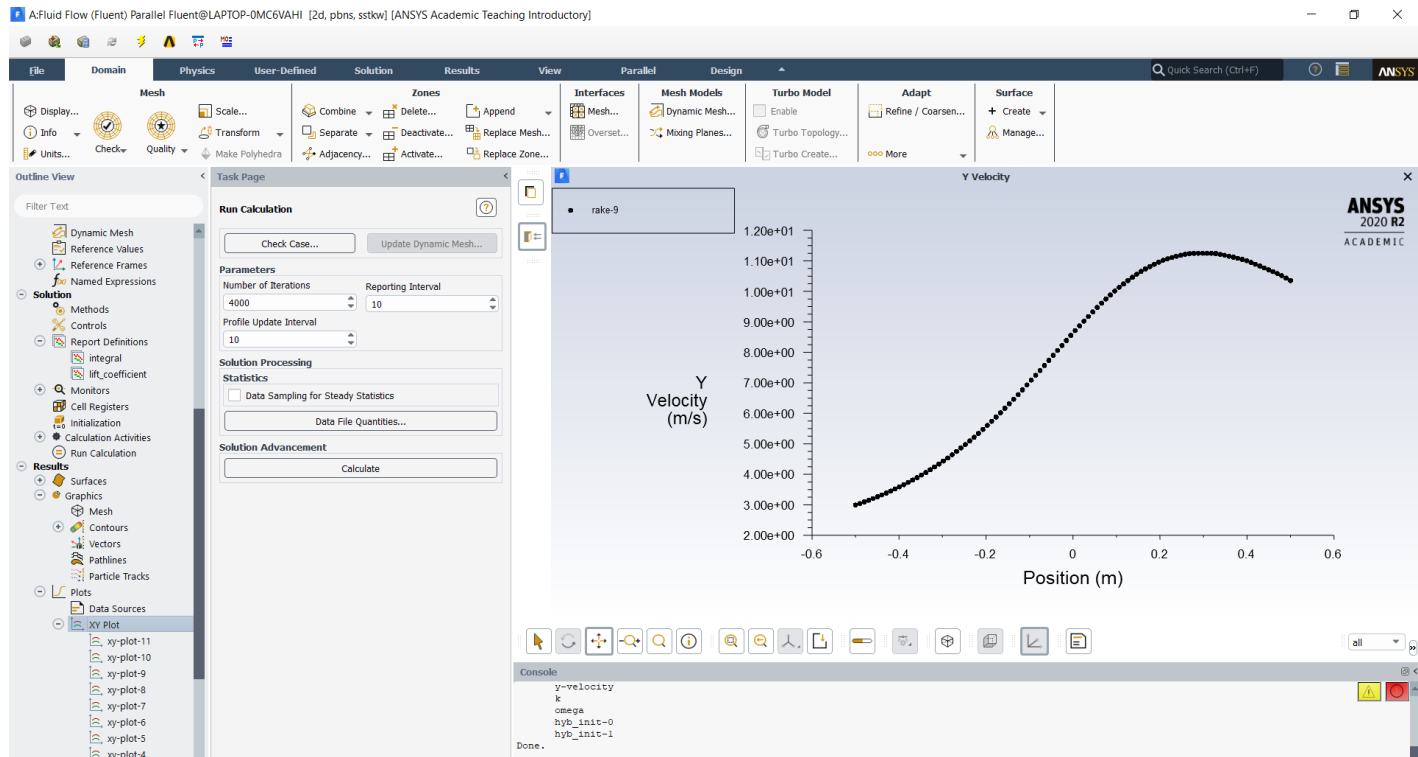
Rake-7



Rake-8



Rake-9



For 3° angle of attack

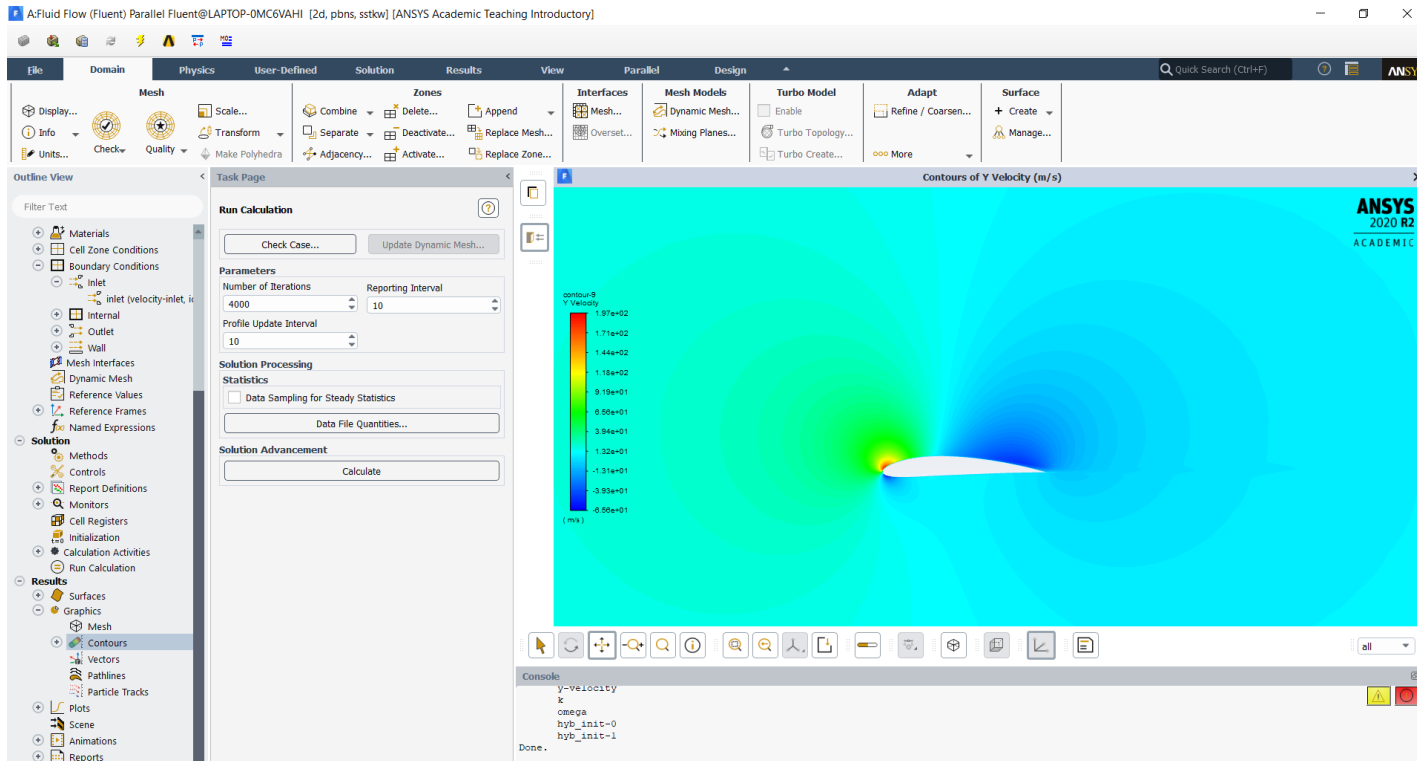
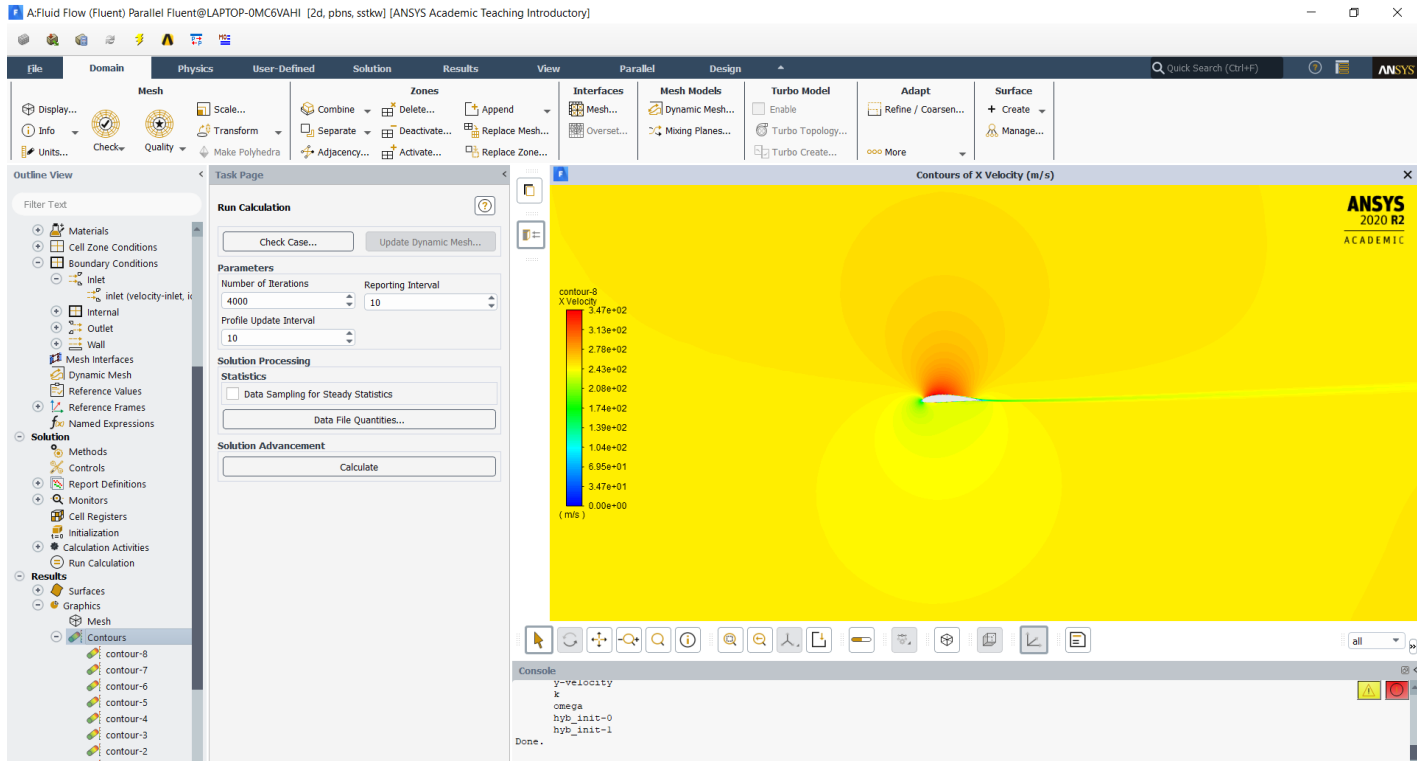
$$\begin{aligned}
 \text{Circulation} &= \int_A^B \mathbf{v} \cdot d\mathbf{l} + \int_B^C \mathbf{v} \cdot d\mathbf{l} + \int_C^D \mathbf{v} \cdot d\mathbf{l} + \int_D^A \mathbf{v} \cdot d\mathbf{l} \\
 &= 541.85 - 0.7 - 480.83 + 28.27 \\
 &= 88.59
 \end{aligned}$$

Air density = 1.225 Kg/m³

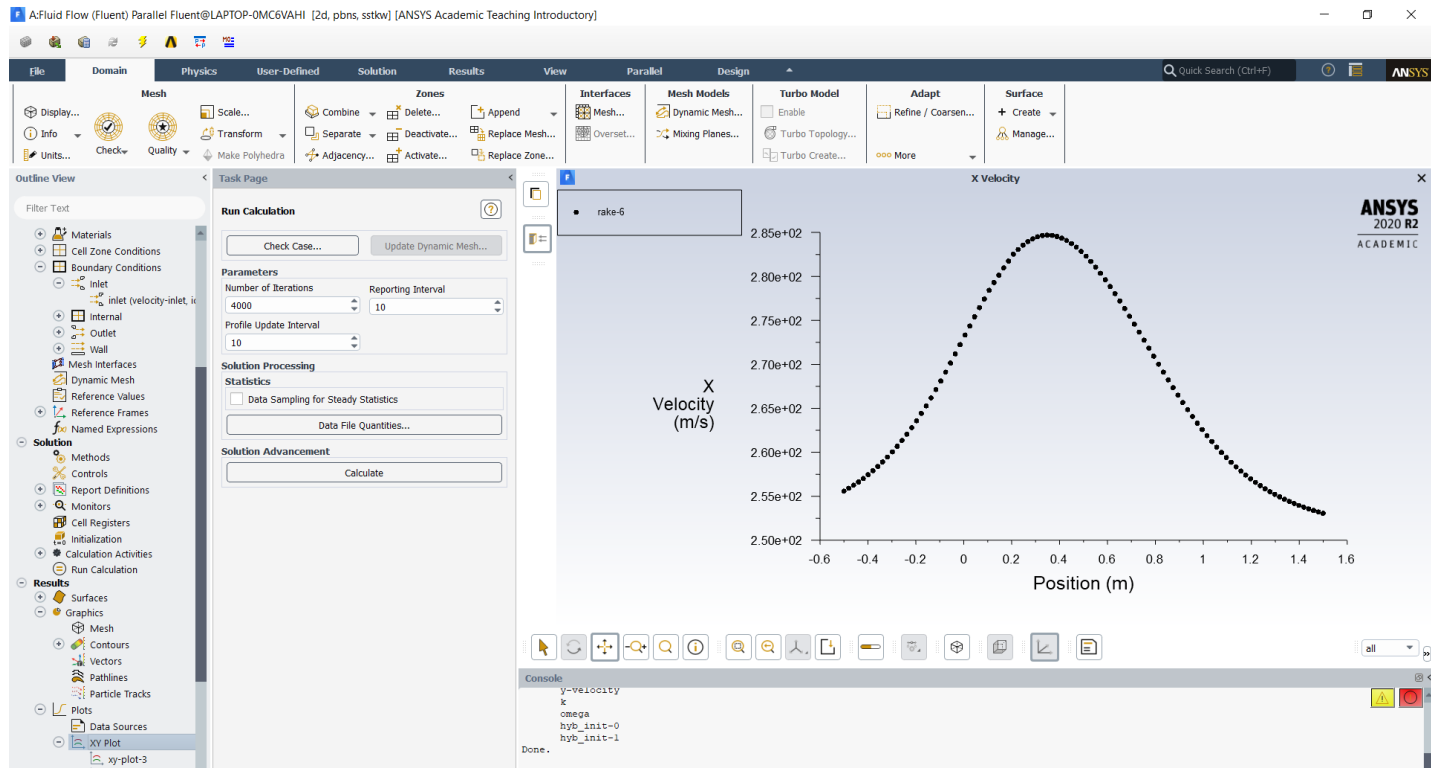
Freestream velocity = 250m/s

Lift = (Air density)*(Freestream velocity)*(circulation)

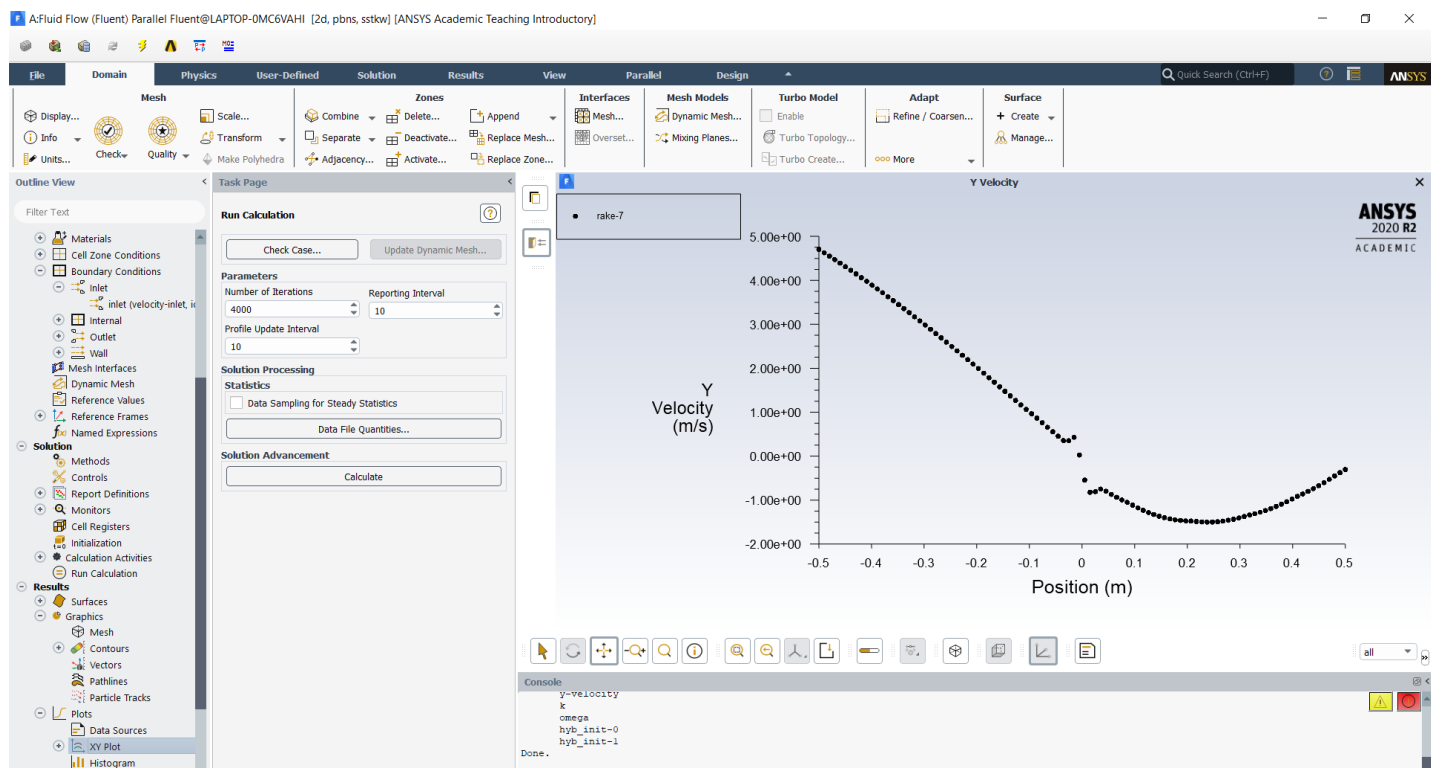
$$= 1.225 * 250 * 88.59 = 27130.69 \text{ N}$$



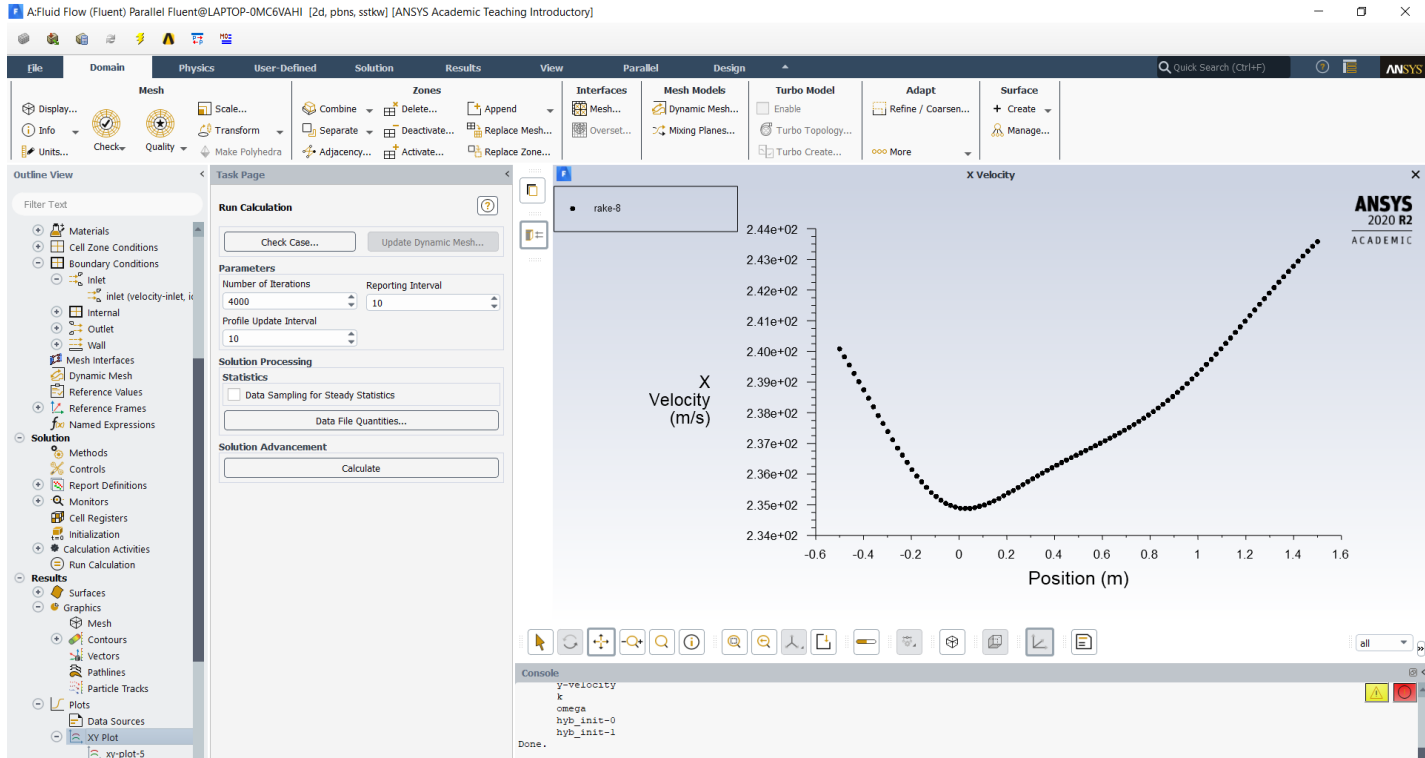
Rake-6



Rake-7



Rake-8



Rake-9

