

General Mission Analysis Tool (GMAT)

User Guide

General Mission Analysis Tool (GMAT): User Guide

2011a

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Chapter 1. Introduction

Introducing GMAT

GMAT is an open-source mission analysis and design tool.

GMAT Interface Design/Philosophy

System Requirements

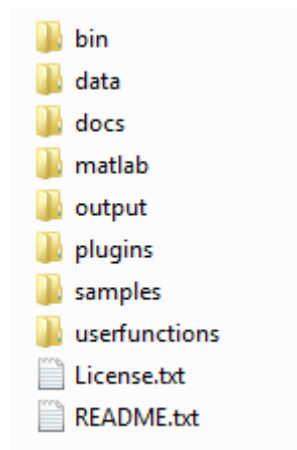
Installation

Data and Configuration

Below we discuss the files and data distributed with GMAT and that are required for GMAT execution. GMAT requires many data files such as planetary ephemeris, Earth orientation data, leap second files and gravity files to name just a few. Below we describe how those files are organized and describe the controls provided so that you can customize the data files GMAT uses at run time.

File Structure

The default directory structure for GMAT is shown below and is broken down into eight main directories. These directories organize the files and data used to run GMAT including binary libraries, data files, texture maps, to 3-D models among many others. The only two files in the GMAT root directory are the license.txt file and the README.txt file. A summary of the contents of each folder is described in further detail in the sections below.



GMAT Directory Structure

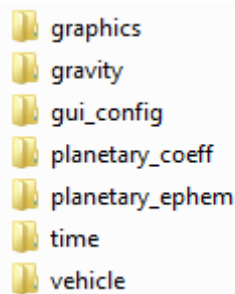
bin Folder

The bin folder contains all binary files required for the core functionality in GMAT (third-party, alpha and beta libraries are placed in the plugins folder). These libraries include the executable file (GMAT.exe on Windows, GMAT.app on Mac, etc.) and libraries for the GUI. The bin folder also contains two text files: gmat_startup_file.txt, and gmat.ini. The startup file is

discussed in detail in a separate section below. The gmat.ini file is used to configure some GUI panels, set paths to external web links, and define "tool tip" messages.

data Folder

The data folder contains all data files required to run GMAT and is organized according to data types as shown in the figure below. The gravity folder contains a folder for each default central body modeled in GMAT and in those folders are files containing gravitational coefficients. The gui_config folder contains files for configuring some of the dialog boxes for GMAT Resources and Commands. These files allow you to custom configure a GUI for a user-provided plugin. Furthermore, some of the built-in dialog box designs employ an ini file for their configuration.



data Folder Structure

The graphics folder contains four subfolders: splash, stars, icons and texture. The splash folder contains the GMAT splash screen that is displayed briefly while GMAT is initializing. The stars folder contains a star catalogue used for displaying stars in 3D graphics. The texture folder contains texture maps used for 3D graphics. The icons folder contains graphics files for icons and images loaded at run time. These include the GMAT logo, images used on the about panel and welcome screen, and icons for the Toolbar, Resource Tree and Mission Tree.

The planetary_coeff folder contains Earth Orientation Parameters (EOP) provided by the International Earth Rotation Service (IERS) and nutation coefficients for different nutation theories. The planetary_ephem folder contains two folders: de and spk. The de folder contains the binary Digital Ephemeris DE405 files for the 8 planets, the Moon, and Pluto developed and distributed by JPL. The spk folder contains an spk kernel built from the DE421 file and kernels for selected comets, asteroids and moons. All ephemeris files distributed with GMAT are in the little-endian representation.

The last two sub-folders in the data folder are time and vehicle. The time folder contains the JPL leap second kernel naif0009.tls and the GMAT leap second file tai-utc.dat. The vehicle folder contains two sub folders: ephemeris and models. The ephemeris subfolder contains SPK ephemeris files - including orbit, attitude, frame, and time kernels - for selected spacecraft. The models folder contains 3D model files.

docs Folder

Documentation for GMAT is contained in the docs folder and includes PDF versions of the User's Guide, Mathematical Specification, Design Specification, and Requirements Specification to name a few. There is also a subfolder named help that contains html help files.

matlab Folder

The matlab folder contains m-files required for GMAT's MATLAB interfaces including the interface to fmincon, and interfaces for driving GMAT from MATLAB. All files in the matlab folder must be included in your MATLAB path for the MATLAB interfaces to function properly.

output Folder

The output folder is the default location for file output such as ephemeris files and report files. If no path information is provided for reports or ephemeris files created during a GMAT session, then those files will be written to the output folder.

plugins Folder

The plugins folder is for third-party libraries and for functionality that is still in alpha or beta status. A "proprietary" sub folder within the plugins directory is for third-party libraries that cannot be distributed as open source files and is an empty folder in the open source distribution.

samples Folder

The samples folder contains many sample missions ranging from Hohmann transfer to Libration point station-keeping, to Mars B-Plane targetting. These files are intended to demonstrate GMAT's capabilities and to provide you with potential starting points for building common mission types for your application and flight regime.

userfunctions Folder

The userfunctions folder contains two subfolders: gmat and matlab. These folders are where gmat and matlab functions are stored that are called in the GMAT command sequences for sample missions distributed with GMAT. You can also store your own custom GMAT and MATLAB functions in these folders.

Configuring GMAT Data Files

You can configure the data files GMAT loads at run time by editing the file named `gmat_startup_file.txt` located in the bin directory. The startup file contains path information to files such as ephemeris, earth orientation data and graphics files among others. By editing the startup file, you can customize which files are loaded and used during a GMAT session. Below we describe the customization features available in the startup file. The order of lines in the startup file does not matter.

Leap Second and EOP files

GMAT reads several files that are used for high fidelity modelling of time and coordinate systems. These files are the leap second files and the Earth Orientation Parameters (EOP) provided by the IERS. The EOP file is updated daily by the IERS. To update your local file with the latest data, simply replace the data in file `eopc04.62-now`, (located in the directory `./data/planetary_coeff/`) with the data provided by the IERS in the link here [http://data.iers.org/products/177/11221/orig/eopc04_05_IAU2000.62-now]. There are two leap second files provided with GMAT. The file named `naif0009.tls` located in the `./data/time` folder is used by the JPL SPICE libraries when computing ephemerides. When a new leap second is added, you can replace your old SPICE leap second file with the new file located here [ftp://naif.jpl.nasa.gov/pub/naif/generic_kernels/lsk/].

GMAT reads the file `tai-utc.dat` in the `./data/time` folder for all time computations requiring leap seconds that are not performed by the SPICE utilities. You can modify this file if a new leap second is added by simply duplicating the last row and updating it with the correct information for the new leap second. For example, if a new leapsecond were added on 01 Jul 2013, then you would add the following line to the `tai-utc.dat` file:

```
2013 JUL 1 =JD 2456474.5 TAI-UTC= 35.0 S + (MJD - 41317.) X 0.0
```


Loading Custom Plugins

Custom plugins are loaded by adding a line to your startup file describing the name and location of the plugin library. In order for a plugin to work with GMAT, the plugin library must be placed in the folder referenced in the startup file. You specify the path to a plugin using the "PLUGIN" keyword and specify the file by providing its name without the file extension (.dll on Windows). For example, to load a Windows plugin named libVF13Optimizer.dll located in the \bin\proprietary folder, you add this line to your startup file:

```
PLUGIN = ./proprietary/libVF13Optimizer
```

User-defined Function Paths

If you create custom GMAT or MATLAB functions for your application, you can provide the path to those files and GMAT will locate them at run time. The default startup file is configured so you can place GMAT files in the ./userfunctions/gmat folder and place MATLAB functions in the ./userfunctions/matlab folder GMAT automatically searches those locations at run time. You can change the location of the search path to your GMAT or MATLAB functions by changing these lines in your startup file to reflect the location of your files with respect to the GMAT bin folder:

```
GMAT_FUNCTION_PATH = ../userfunctions/gmat
MATLAB_FUNCTION_PATH = ../userfunctions/matlab
```

If you wish to organize your custom functions in multiple folders, you can add multiple search paths to the startup file. For example,

```
GMAT_FUNCTION_PATH = ../MyFunctions/utils
GMAT_FUNCTION_PATH = ../MyFunctions/StateConversion
GMAT_FUNCTION_PATH = ../MyFunctions/TimeConversion
```

Configuring the MATLAB Interfaces

GMAT supports several MATLAB interfaces and to use these interfaces the following MATLAB folders must be added to your system path:

```
MATLAB/bin/win32
MATLAB/bin
MATLAB/runtime/win32
```

Caution

The above folders are added to your system path during MATLAB installation. However, for some versions of MATLAB (2010a for example), MATLAB and Windows are distributed with libraries that have the same name. This may cause the Windows libraries to load instead of the MATLAB libraries. As a result, you may need to put the folders above at the beginning of your system path.

If you have multiple versions of MATLAB installed, GMAT will use the version that appears first in your system path and that version must be registered as a COM server using the MATLAB regserver command. To register the desired version of MATLAB, using a command

prompt change directory to the bin folder of the desired MATLAB release and run the following command.

```
matlab.exe -regserver
```

Finally you must add to your MATLAB path all files in the matlab directory which is located in the GMAT root directory.

Support and Resources

Chapter 2. How To

Reporting mission parameters

Running GMAT Scripts from MATLAB

Overview

GMAT was designed to allow users to run GMAT scripts through MATLAB®. This feature gives the user greater control and flexibility of GMAT that cannot be done with just the GMAT to MATLAB® interface or GMAT alone. For example, if a user would like to dynamically change scripts and run them, that can currently only be done using the MATLAB® to GMAT interface. A MATLAB® script can also be generated to run GMAT scripts that are located in multiple folders.

If, after running through this tutorial, you still have difficulties with the interfaces between MATLAB® and GMAT working, visit our MATLAB Help Forum [<http://gmat.ed-pages.com/forum/viewforum.php?f=14>] or our MATLAB<->GMAT Interface FAQ [<http://gmat.ed-pages.com/forum/viewtopic.php?f=14&t=65>] forum topic.

Files and Folders to be used

- [rootGMATpath] - folder with the GMAT executable
- Ex_TargetHohmannTransfer.script - Sample Mission script used for running GMAT from Matlab
- runMatlabToGMAT.m [scripts/runMatlabToGMAT.m] - MATLAB® script that runs the GMAT script
- runMatlabToGMATsimple.m [scripts/runMatlabToGMATsimple.m] - simplified MATLAB® script that runs the GMAT script

Procedure

1. Place the downloaded script runMatlabToGMAT.m script into the [rootGMATpath]\matlab\ directory

WARNING! This script clears the MATLAB® workspace. Save your current MATLAB® data if it's needed

2. Make sure the script Ex_TargetHohmannTransfer.script is in the [rootGMATpath]\input\SampleMissions\ directory
3. Unless it has been done already, open MATLAB® and add [rootGMATpath]\matlab\ directory and the sub-directories to the MATLAB® path.
4. Open GMAT and start the Matlab Server.

*The server can be started by navigating in the Resources tree to the Interfaces folder. Right click the Matlab Server object and select Start.

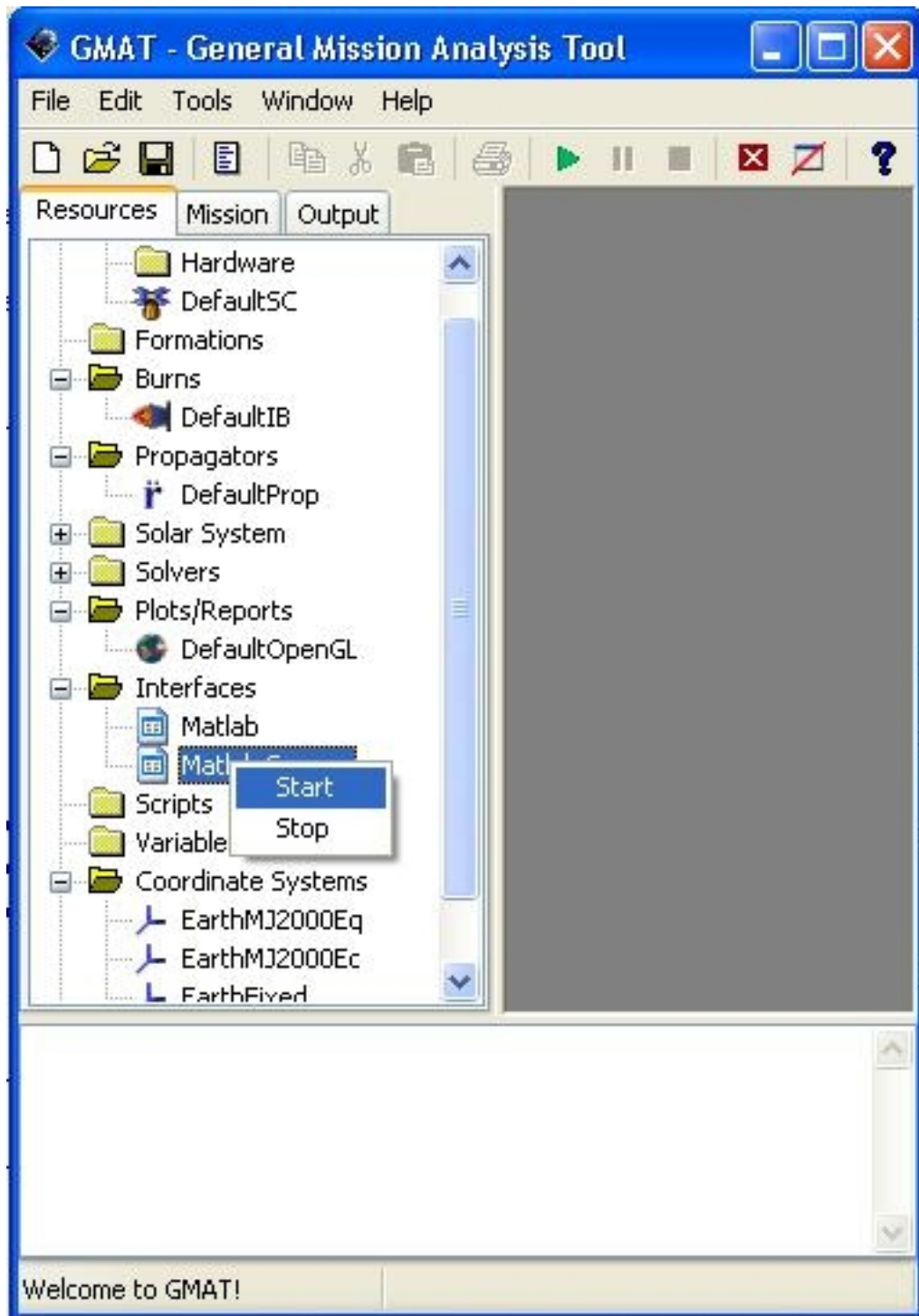


Figure: Opening Matlab Server in GMAT Resource Tree

5. Run runMatlabToGMAT.m in MATLAB®
6. Open a file browser to the [rootGMATpath]\output\SampleMissions\ directory and open the ASCII Ex_TargetHohmannTransfer.report script in a text editor.
7. Scroll down to the last few lines, and notice the extra lines indicating that this script was run from MATLAB® . If the Ex_TargetHohmannTransfer.script file is run from GMAT, these lines will not be present.

8. Congratulations - you have finished the main section of this tutorial. Be sure to open the runMatlabToGMAT.m script, to understand how GMAT is controlled by MATLAB®. This MATLAB® script is heavily commented to explain what is being done. If you still have questions, feel free to post them at our GMAT Source Forge forums (<http://sourceforge.net/projects/gmat/>)

Additional Instructions

The runMatlabToGMAT.m Matlab script might be too complicated for novice MATLAB® users. If so, use a simplified version of the script, runMatlabToGMATsimple.m. For this script a .m file is needed, so copy the Ex_TargetHohmannTransfer.script and rename the copy as Ex_TargetHohmannTransfer.m. Follow the above instructions but first rename the runMatlabToGMAT.m file to runMatlabToGMATsimple.m.

Creating ephemeris files

Creating a Report

Objective and Overview

The objective of this tutorial is to demonstrate how to report parameters out to a file using two different reporting techniques, as well as how to use strings to improve the readability of a report file.

Download the script file: ReportFile.script [scripts/ReportTutorial.script]

Prerequisites

- Objects modified (use if you have trouble with this tutorial):
 - ReportFile
- Commands modified (use if you have trouble with this tutorial):
 - Report Command
 - ScriptEvent Command

Mission Description

- Objective: Use the Report Object to output parameters to an ascii file
- Assume: N/A
- Find: N/A

Resource, Mission, and Output Trees

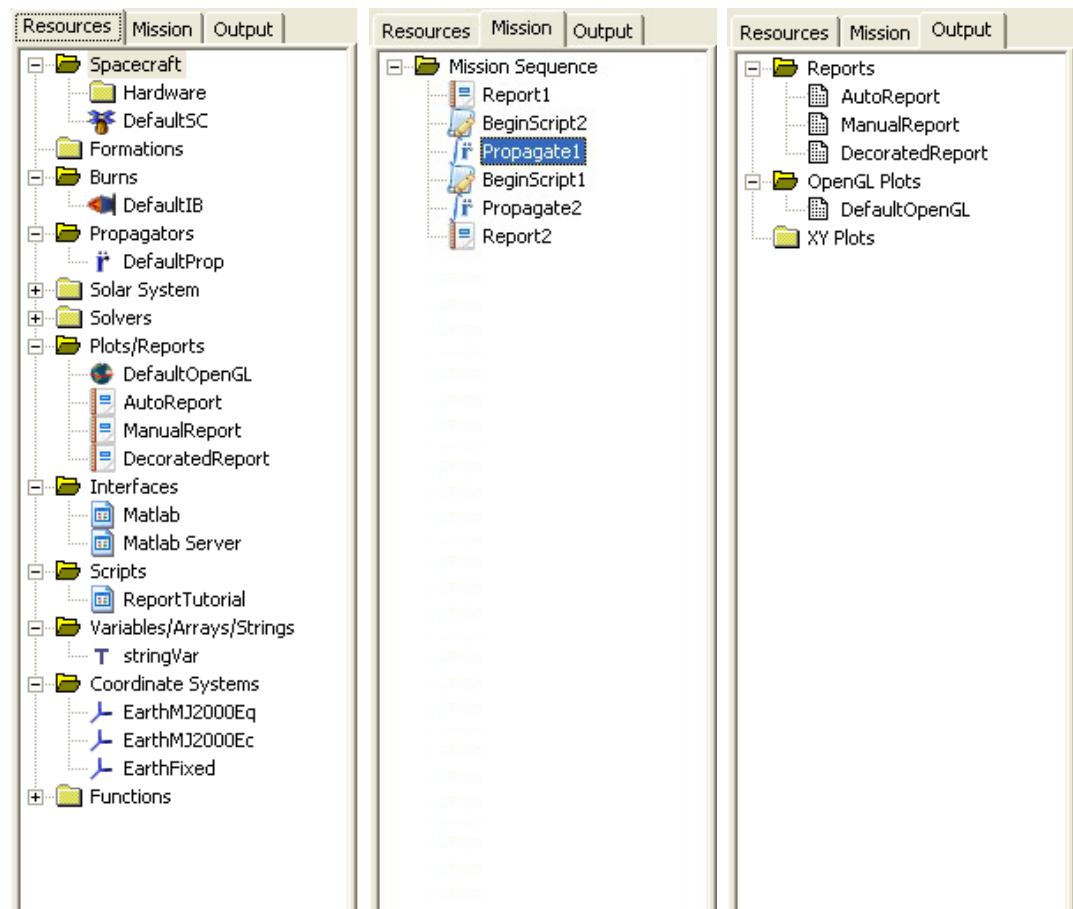


Figure: Creating a Report Resource, Mission, and Output Trees

Creating and Configuring the Resource Tree

Objects Required

- Plots/Reports:
 - ReportFile - AutoReport, ManualReport, DecoratedReport
- Variables/Arrays/Strings:
 - String Variable - StringVar

Creating and Modifying Objects

Add ReportFile Objects

The ReportFile has two features that both allow a user to output data to an ASCII text file. One feature of a ReportFile is to output data at every integrator step and the other is to output data at the user's discretion using Report commands in the Mission Sequence.

- Right click on the Plots/Reports folder and add a ReportFile.
- Add two more ReportFiles for a total of three ReportFiles.
- Rename the three ReportFiles by right clicking the ReportFile and selecting Rename. The new ReportFile names are as followed: AutoReport, ManualReport, and DecoratedReport.

- Verify that your Plots/Reports Resource Tree settings are identical to the image below.

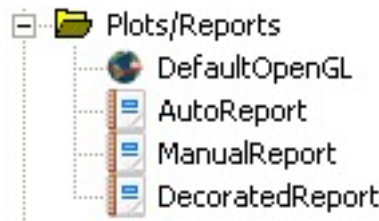


Figure: Plots/Reports Resource Tree Folder

Configure AutoReport ReportFile

The AutoReport ReportFile is used to report data every integrator step size once the mission is run.

- Open the properties panel of the AutoReport ReportFile and remove all parameters in the Selected input field by clicking the Remove All button.
- Add the following parameters to the Selected input field in the order specified: DefaultSC.UTCGregorian, DefaultSC.EarthMJ2000Eq.X, DefaultSC.EarthMJ2000Eq.Y, DefaultSC.EarthMJ2000Eq.Z, DefaultSC.EarthMJ2000Eq.VX, DefaultSC.EarthMJ2000Eq.VY, DefaultSC.EarthMJ2000Eq.VZ
- Change the File input field to ./output/AutoReport.txt .
- Verify that your settings are identical to the image below.

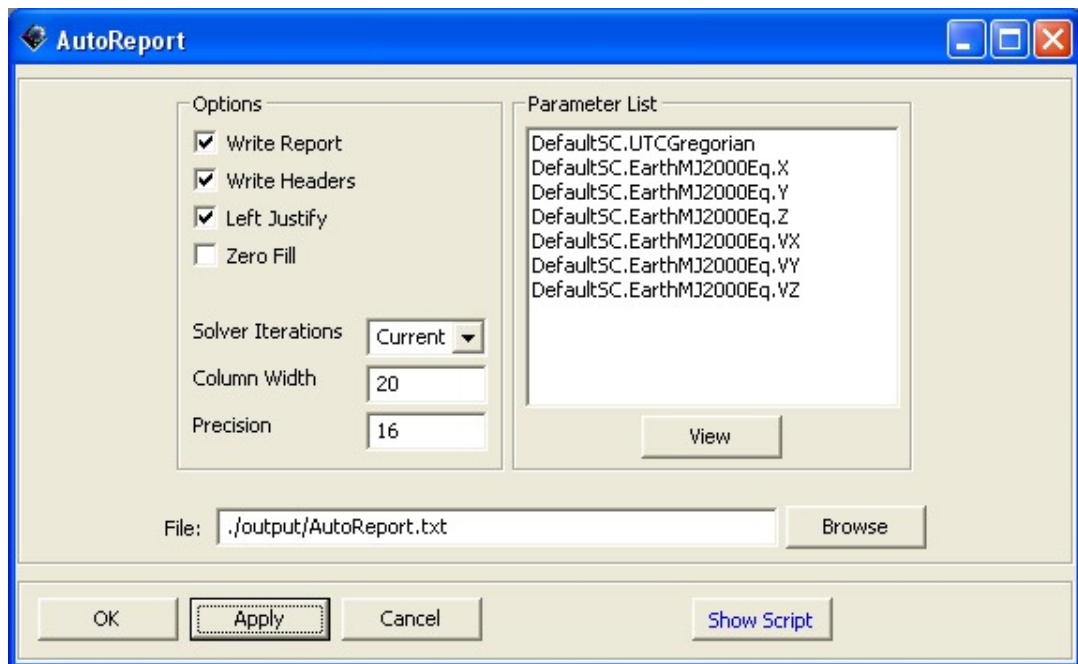


Figure: AutoReport ReportFile Settings

Configure ManualReport ReportFile

- Open the properties panel of the ManualReport ReportFile
- Remove all parameters in the Selected input field by clicking the Remove All button.
- Change the File input field to ./output/ManualReport.txt .
- Verify that your settings are identical to the image below.

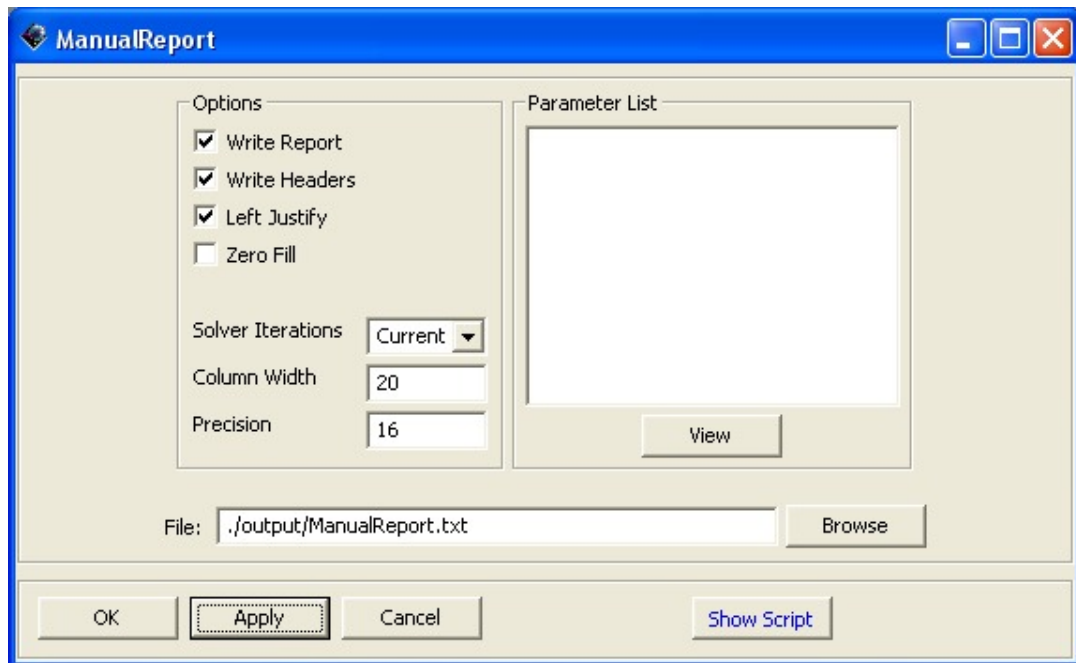


Figure: ManualReport ReportFile Settings

Configure DecoratedReport ReportFile

The DecoratedReport ReportFile is identical to AutoReport ReportFile, besides the file path of the report.

- Open the properties panel of the DecoratedReport ReportFile
- Remove all parameters in the Selected input field by clicking the Remove All button.
- Add the following parameters to the Selected input field in the order specified: DefaultSC.UTCGregorian, DefaultSC.EarthMJ2000Eq.X, DefaultSC.EarthMJ2000Eq.Y, DefaultSC.EarthMJ2000Eq.Z, DefaultSC.EarthMJ2000Eq.VX, DefaultSC.EarthMJ2000Eq.VY, DefaultSC.EarthMJ2000Eq.VZ
- Change the File input field to ./output/DecoratedReport.txt .

Add a String Variable for Decorated Report

- Right click the Variables/Arrays folder and select Add --> String.
- Navigate to the String group box and enter a string Name of stringVar.
- Leave the Value input field blank.
- Click the Create button and click OK to exit out of that panel.
- Verify that your settings are identical to the image below.

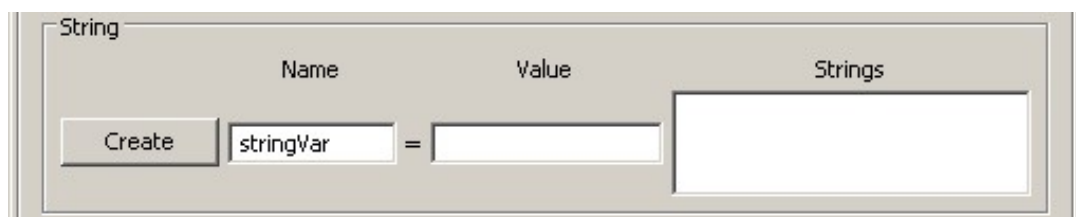


Figure: String Variable Creation

Creating and Configuring the Mission Tree

Add Misc. Commands

- Select the Mission tab in the left pane to switch over to the Mission Sequence tree.
- Right click on the Propagate1 command object and Insert After a ScriptEvent.
- Right click on the newly created ScriptEvent(BeginScript) and Insert After a Propagate.
- Verify that your settings are identical to the image below.

(Folder icons and command numbers may not be the same)



Figure: Mission Tree Configuration 1

Add Report Commands for Manual Report

For the ManualReport ReportFile, we will add a Report command before and after the entire mission sequence to see what the initial state and final state is.

- Right click the first Propagate command and Insert Before a Report command.
- Add the same parameters as the (Configure AutoReport ReportFile):

DefaultSC.UTCGregorian, DefaultSC.EarthMJ2000Eq.X, DefaultSC.EarthMJ2000Eq.Y,
 DefaultSC.EarthMJ2000Eq.Z, DefaultSC.EarthMJ2000Eq.VX,
 DefaultSC.EarthMJ2000Eq.VY, DefaultSC.EarthMJ2000Eq.VZ

- Select the ManualReport ReportFile before closing the Report command properties panel.
- Right click the last Propagate command and Insert After a Report command.
- Add the same parameters as above and select ManualReport for the ReportFile.
- Verify that your settings are identical to the image below.

(Folder icons and command numbers may not be the same)

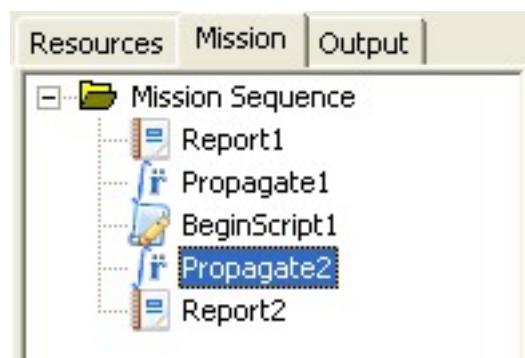


Figure: Mission Tree Configuration 2

Output String Values for Decorated Report

- Right click the first Propagate command and Insert Before a ScriptEvent command.
- Open the properties panel for the first ScriptEvent
- Enter the following between the BeginScript/EndScript text and click OK to exit the panel:

```
GMAT DecoratedReport.WriteHeaders = Off;
stringVar = '=====';
Report DecoratedReport stringVar;
stringVar = GMAT Report File in Time X Y Z VX VY VZ format;
Report DecoratedReport stringVar;
stringVar = '=====';
Report DecoratedReport stringVar;
GMAT DecoratedReport.WriteHeaders = On;
```

Observe how WriteHeaders was turned off while outputting string data. If you don't do this, you will get the word stringVar showing up above each report output line.

- Verify that your settings are identical to the image below.

(Folder icons and command numbers may not be the same)

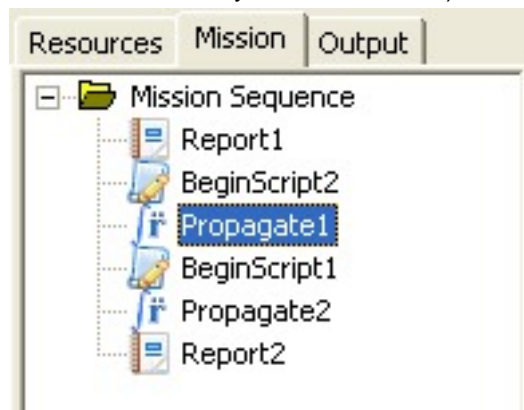


Figure: Mission Tree Configuration 3

This concludes the configuration of the commands needed for the Mission Sequence.

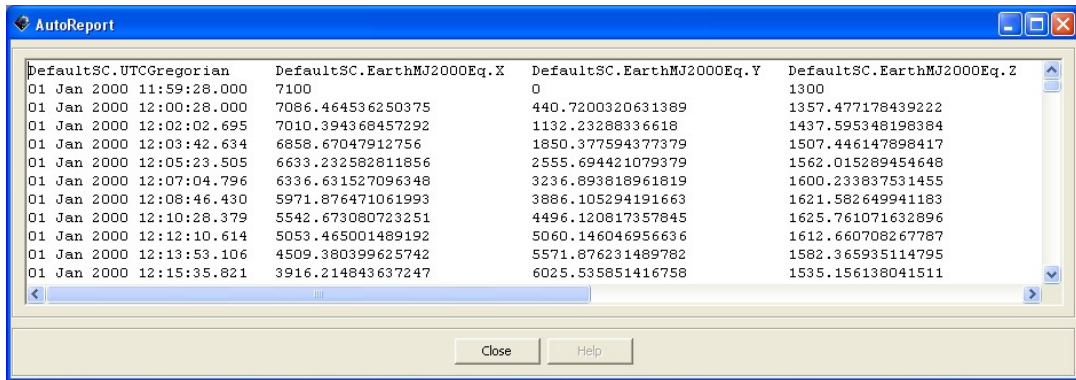
Running the Mission

Run the mission and navigate to the Output Tab.

AutoReport Output

Open the AutoReport file. The contents are a heading and data associated with the parameters in the heading at each propagator integration step.

The AutoReport output should look like the image below.



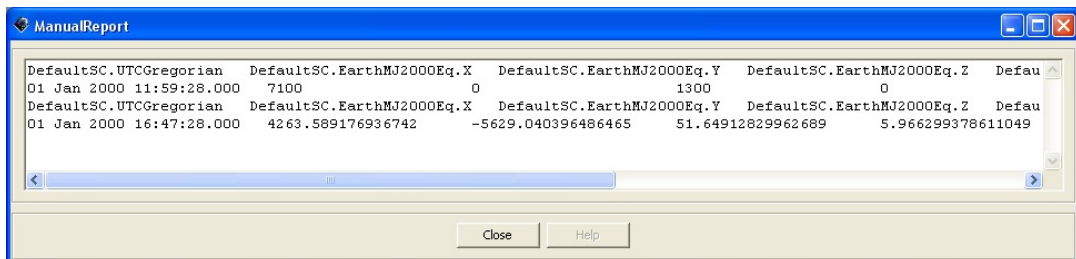
DefaultSC.UTCGregorian	DefaultSC.EarthMJ2000Eq.X	DefaultSC.EarthMJ2000Eq.Y	DefaultSC.EarthMJ2000Eq.Z
01 Jan 2000 11:59:28.000	7100	0	1300
01 Jan 2000 12:00:28.000	7086.464536250375	440.7200320631389	1357.477178439222
01 Jan 2000 12:02:02.695	7010.394368457292	1132.23288336618	1437.595348198384
01 Jan 2000 12:03:42.634	6858.67047912756	1850.377594377379	1507.446147898417
01 Jan 2000 12:05:23.505	6633.232582811856	2555.694421079379	1562.015289454648
01 Jan 2000 12:07:04.796	6336.631527096348	3236.893818961819	1600.233837531455
01 Jan 2000 12:08:46.430	5971.876471061993	3886.105294191663	1621.582649941183
01 Jan 2000 12:10:28.379	5542.673080723251	4496.120817357845	1625.761071632896
01 Jan 2000 12:12:10.614	5053.465001489192	5060.146046956636	1612.660708267787
01 Jan 2000 12:13:53.106	4509.380399625742	5571.876231489782	1582.365935114795
01 Jan 2000 12:15:35.821	3916.214843637247	6025.535851416758	1535.156138041511

Figure: AutoReport Output Results

ManualReport Output

Open the ManualReport file. The contents are a heading followed by the initial spacecraft state, a heading, and the final spacecraft state.

The ManualReport output should look like the image below.



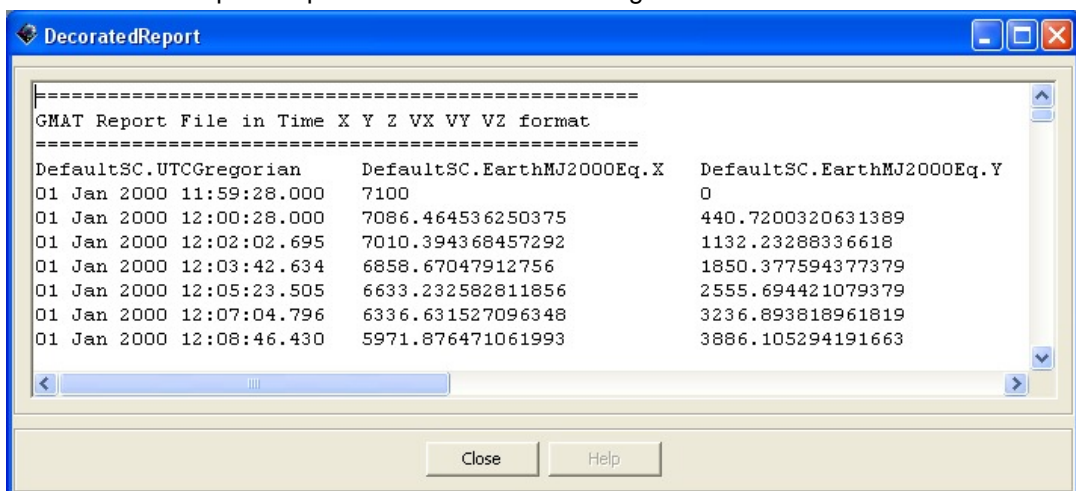
DefaultSC.UTCGregorian	DefaultSC.EarthMJ2000Eq.X	DefaultSC.EarthMJ2000Eq.Y	DefaultSC.EarthMJ2000Eq.Z	DefaultSC.EarthMJ2000Eq.VX	DefaultSC.EarthMJ2000Eq.VY	DefaultSC.EarthMJ2000Eq.VZ
01 Jan 2000 11:59:28.000	7100	0	1300	0	0	0
01 Jan 2000 16:47:28.000	4263.589176936742	-5629.040396486465	51.64912829962689	5.966299378611049	0	0

Figure: ManualReport Output Results

DecoratedReport Output

Open the DecoratedReport file. The contents are the custom heading using string variables, followed by the default heading, and data associated with the parameters in the default heading at each propagator integration step.

The DecoratedReport output should look like the image below.



=====						
GMAT Report File in Time X Y Z VX VY VZ format						
=====						
DefaultSC.UTCGregorian	DefaultSC.EarthMJ2000Eq.X	DefaultSC.EarthMJ2000Eq.Y	DefaultSC.EarthMJ2000Eq.Z	DefaultSC.EarthMJ2000Eq.VX	DefaultSC.EarthMJ2000Eq.VY	DefaultSC.EarthMJ2000Eq.VZ
01 Jan 2000 11:59:28.000	7100	0	1300	0	0	0
01 Jan 2000 12:00:28.000	7086.464536250375	440.7200320631389	1357.477178439222	0	0	0
01 Jan 2000 12:02:02.695	7010.394368457292	1132.23288336618	1437.595348198384	0	0	0
01 Jan 2000 12:03:42.634	6858.67047912756	1850.377594377379	1507.446147898417	0	0	0
01 Jan 2000 12:05:23.505	6633.232582811856	2555.694421079379	1562.015289454648	0	0	0
01 Jan 2000 12:07:04.796	6336.631527096348	3236.893818961819	1600.233837531455	0	0	0
01 Jan 2000 12:08:46.430	5971.876471061993	3886.105294191663	1621.582649941183	0	0	0

Figure: DecoratedReport Output Results

Visualizing a trajectory

Chapter 3. Samples and Tutorials

Propagating a Spacecraft

Objective and Overview

The objective of this tutorial is to teach you how to create a spacecraft and a propagator, and then propagate the spacecraft to orbit perigee by following these basic steps:

1. Create a spacecraft and set its epoch and orbital elements.
2. Create and configure a propagator.
3. Modify the default Orbit View to visualize the trajectory.
4. Configure the mission sequence to propagate the spacecraft to periapsis.

Configuring Resources

Creating and Configuring a Spacecraft

In this section, you'll learn how to set a spacecraft's initial epoch and classical orbital elements. You'll need GMAT open with the default mission loaded. The default mission is loaded when a new session of the GMAT executable is started or when the New Mission button in the Toolbar is clicked.

Creating a Spacecraft

Working from the GUI, you can create a new spacecraft by starting at the Resource Tree.

1. Right click on the Spacecraft folder, and select Add Spacecraft
2. Rename the spacecraft by right-clicking on the new spacecraft and selecting Rename from the drop down menu. For this tutorial, name the Spacecraft "Sat".

Setting a Spacecraft's Epoch

1. Double left click on the Spacecraft icon for the new spacecraft Sat in the Resource Tree to open the Spacecraft's dialog box. If it is not already selected, click on the Orbit Tab.
2. Left-click the Epoch Format drop-down menu and select UTCGregorian. You'll see the value in the Epoch field change to the UTC Gregorian epoch format.
3. Left-click in the Epoch field, and type the desired value of 22 Jul 2014 11:29:10.811 (or you can cut and paste from the text in this tutorial).
4. Save the changes by left clicking the Apply button at the bottom of the window. In the first figure below you see the orbit tab after correctly setting the epoch to the desired value.

Setting a Spacecraft's State

We'll use the Keplerian orbital elements for this tutorial and we'll enter them with respect to Earth's MJ2000 Equator system.

1. Open the spacecraft dialog box for the spacecraft named Sat if it is not already open. (See section named Setting a Spacecraft's Epoch just above if you don't remember how to do this).

2. Change the state type to Keplerian by left-clicking on the State Type drop down menu and selecting "Keplerian". You will see the GUI reconfigure to the Keplerian representation.
3. Set the values in the GUI to those seen in the image below.
4. Compare the GUI to the screen capture below to ensure you have completed the steps correctly.
5. Save the changes and close the dialog box by pressing the OK button.
6. Save your work to the hard disk, press the save button located in the toolbar. If this is the first time you've saved the mission, you'll be prompted to provide a name and location for the file.

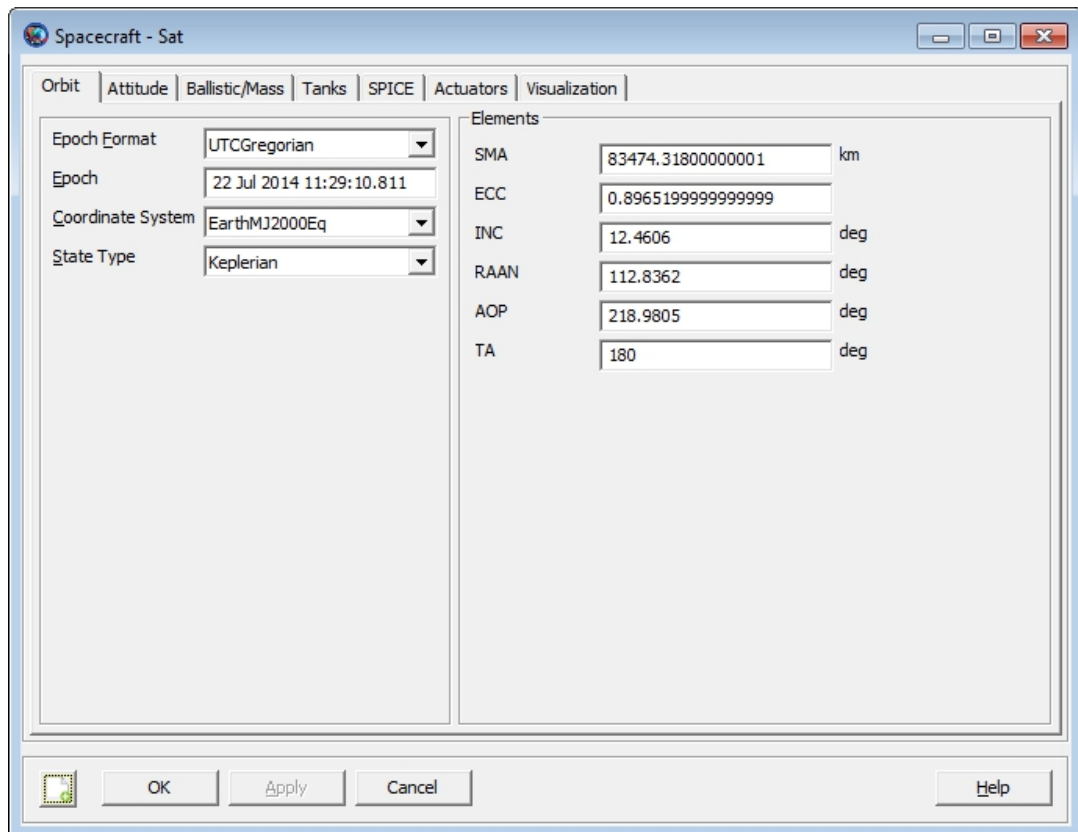


Figure: Spacecraft State Setup

Creating and Configuring a Propagator

In this section you'll learn how to create a new propagator from the GUI. This involves selecting a numerical integrator and a force model. Let's start by creating a new propagator from the GUI.

Creating a Propagator

1. Locate the Propagators folder in the Resource Tree.
2. Right-click on the Propagators folder and select Add Propagator from the drop-down menu.
3. Rename Propagator1 as "LowEarthProp". To do this, right click on the newly created Propagator1 and select Rename. In the dialog box that appears, type LowEarthProp and hit OK.

Look at the dialog box for LowEarthProp by double left clicking on its icon under the Propagators folder. On the left side of the propagator dialog box you see where you can select the desired numerical integrator and configure it for your application. On the right hand side of the panel are combo boxes and lists that allow the user to set up the force model. Now let's look at how to configure a force model.

Configuring a Force Model

For this tutorial we will use an Earth 10x10 non-spherical gravity model, Jacchia-Roberts atmospheric model, and point mass perturbations from the Sun and Moon.

1. Open LowEarthProp from the Propagators folder in the Resource tree
2. Locate the Primary Bodies group on the Propagator dialog box. In the Gravity group box, change the degree and order to 10 by left clicking in the text field and typing in the values.
3. Locate the Atmosphere Model pull-down menu in the Drag group.
4. Left click on the pull-down menu and select JacchiaRoberts. (For now we will leave the default options for Jacchia-Roberts model)
5. Left click on the Select button to the right of the Point Masses text box input field, which brings up a list box containing all gravitational bodies that have not been included in the Primary Bodies list (See Figure below).
6. Left click on "Sun" under the available bodies list, and then left click the right arrow found between the available bodies and Selected Bodies list boxes.
7. Add the Moon (named Luna in GMAT) following the same procedure as the preceding step.
8. Left click on the OK button to save changes.

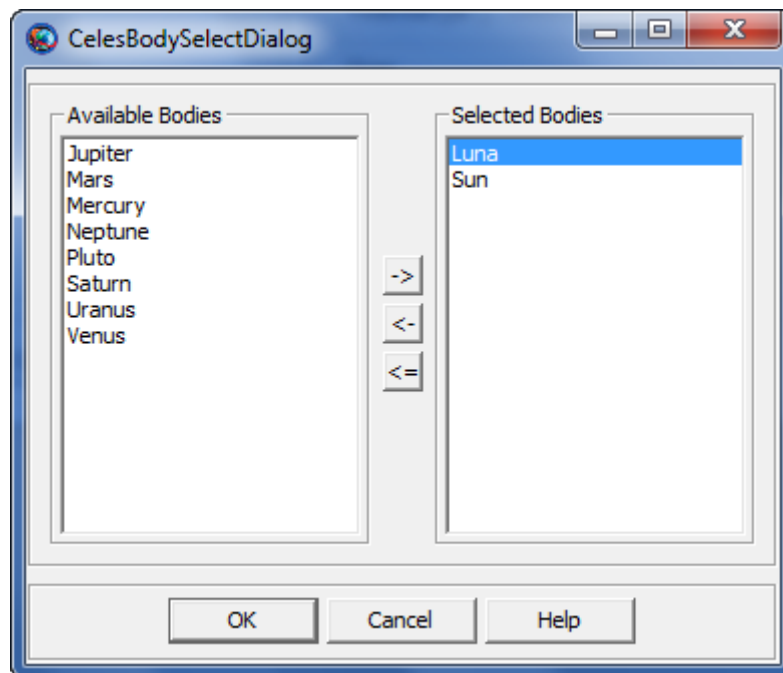


Figure: Force Model Point Mass Configuration

Below is an illustration after correctly configuring the force model according to the instructions above.

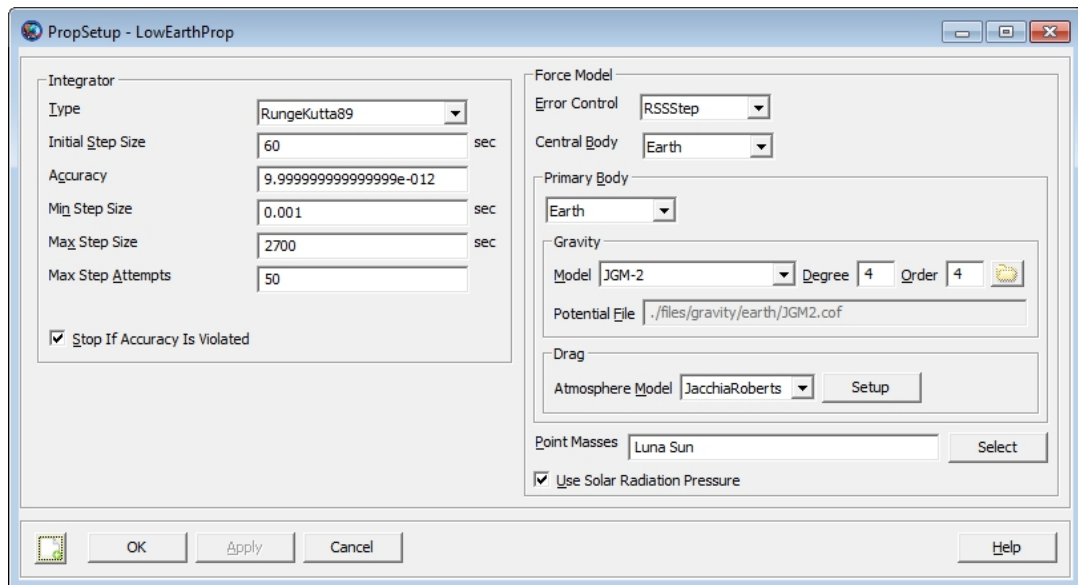


Figure: Force Model Configuration

Configuring the Default Orbit View Plot

In this section, we'll configure the default Orbit View plot to show the spacecraft we've created above. We'll remove DefaultSC from the list of objects to appear in the plot, add Sat, and change the view point so we can see the entire orbit when we propagate the spacecraft.

1. Locate the Output folder under the Resource Tree and double-left-click on DefaultOrbitView to open its dialog box.
2. Locate the View Object group, and find the Selected Spacecraft list.
3. Left click on DefaultSC under Selected Spacecraft and then click the left arrow button that appears to the left of the Selected Spacecraft list. This removes DefaultSC from the plot.
4. Locate the Spacecraft list in the View Object group.
5. Left click on Sat, and then left click on the right-pointing arrow button that appears to the right of the Spacecraft list.

The orbit for Sat is a highly eccentric orbit, and to view the entire orbit, we need to change the settings in the View Definition group.

1. Locate the View Point Vector settings in the View Definition group.
2. In the text boxes to the right of the ViewPointVector pull-down menu, enter 30000, -5000, and 5000 respectively as shown in illustration below
3. Uncheck the DrawXY Plane box located in the Drawing Options group.

See the figure below for the correct configuration of DefaultOrbitView after completing the steps above. When you are finished click Ok to save the changes.

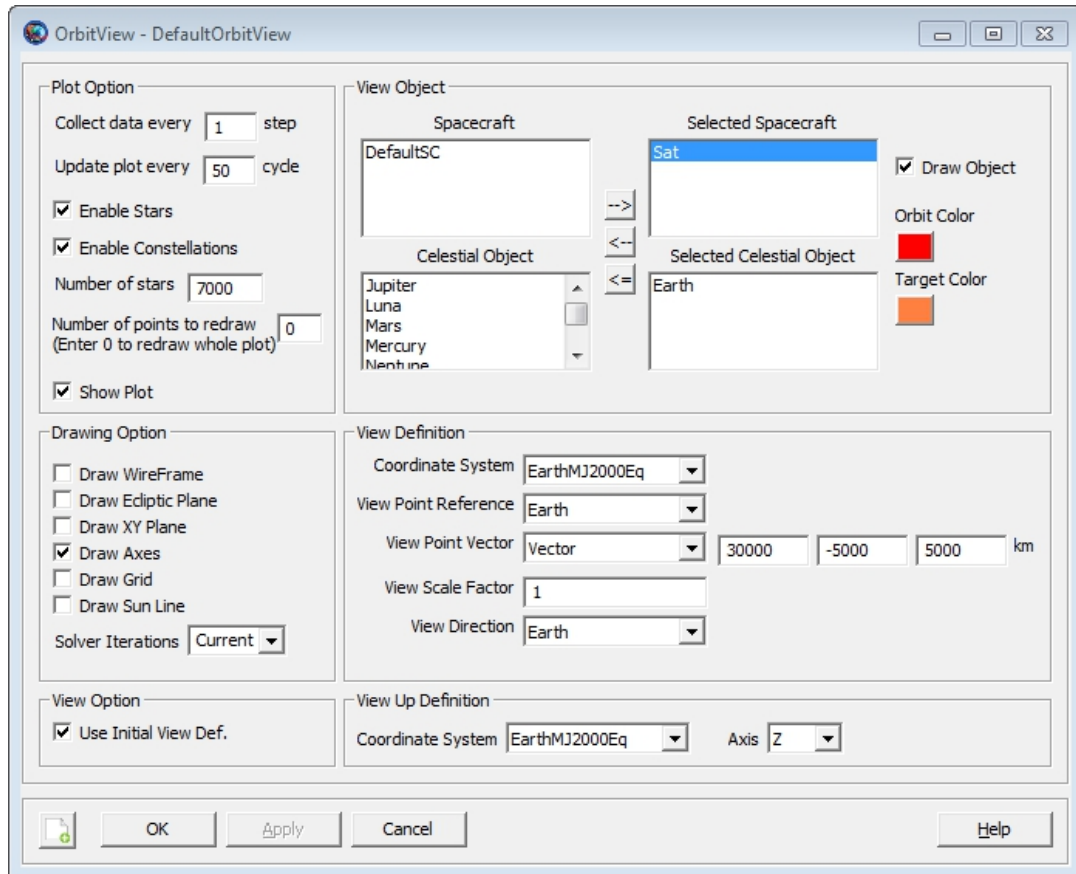


Figure: DefaultOrbitView Configuration

Configuring the Mission Tree

Creating and Configuring a Propagate Command

Creating a Propagate Command

The last step in this tutorial is to propagate the spacecraft. To do this, we'll configure a propagate command in the mission and configure it to propagate until the spacecraft reaches orbit periapsis.

1. Locate the Mission tab and left click on it to bring up the mission sequence. The Mission tab is located next to the Resource tab located near the upper left hand corner of the GMAT GUI.
2. Double left click on the event named Propagate1. You will see a dialog box that allows you to configure the command.

Selecting a Propagator

1. Locate the propagator list in the Propagators and Spacecraft group box.
2. Right click on the field where you see DefaultProp.
3. Left click on LowEarthProp and hit Ok.

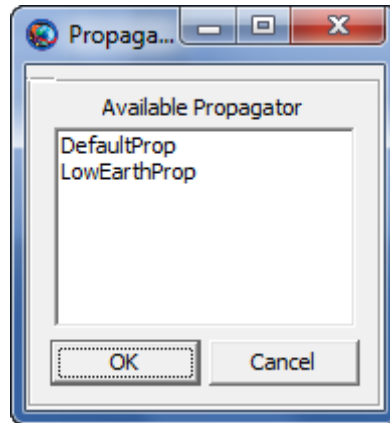


Figure: Propagate Command Propagator Select Dialog Configuration

Selecting a Spacecraft

To propagate the new spacecraft named Sat, we need to add it to the Spacecraft List group where originally you see DefaultSC. To propagate the Spacecraft named Sat:

1. Right-click in the field where you see DefaultSC.
2. Remove DefaultSC from the list by left clicking on it and then left-click on the left arrow button.
3. Left click on Sat and then left click on the right arrow button to add it to the list of spacecraft to propagate.
4. Left click on the OK button.

Below you see the dialog box after correctly configuring it to propagate the spacecraft named Sat.

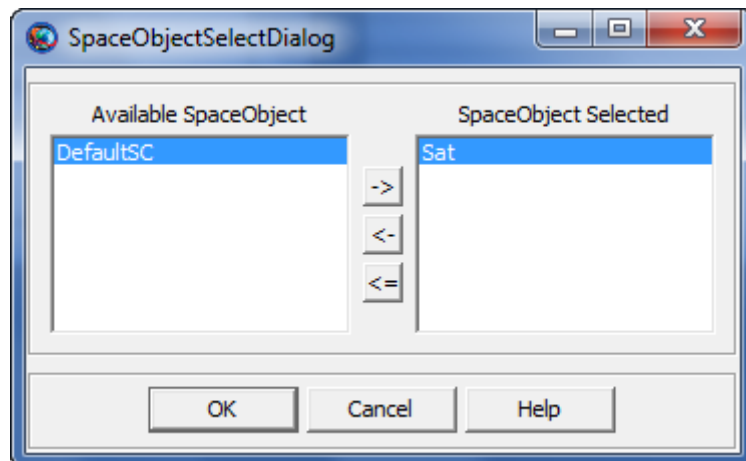


Figure: Propagate Command Space Object Select Dialog Configuration

Selecting a Stopping Condition

The only remaining thing to do in configuring the propagate event, is to tell the propagator when to stop the propagation.

1. Locate the Stopping Conditions group, and right click in the field that currently says DefaultSC.ElapsedSecs located in the Parameter column.
2. Find the Object List and choose Sat instead of Default Sat. (This tells GMAT the stopping condition will be associated with the spacecraft named Sat.)

3. Locate the Property list in the middle of the dialog box and scroll down to locate Periapsis.
4. Left click on the word Periapsis, and then left click on the right-pointing arrow found in the center column of the dialog box.

The dialog box below shows an example after correctly performing these instructions. Once you are finished, click Ok to save your changes.

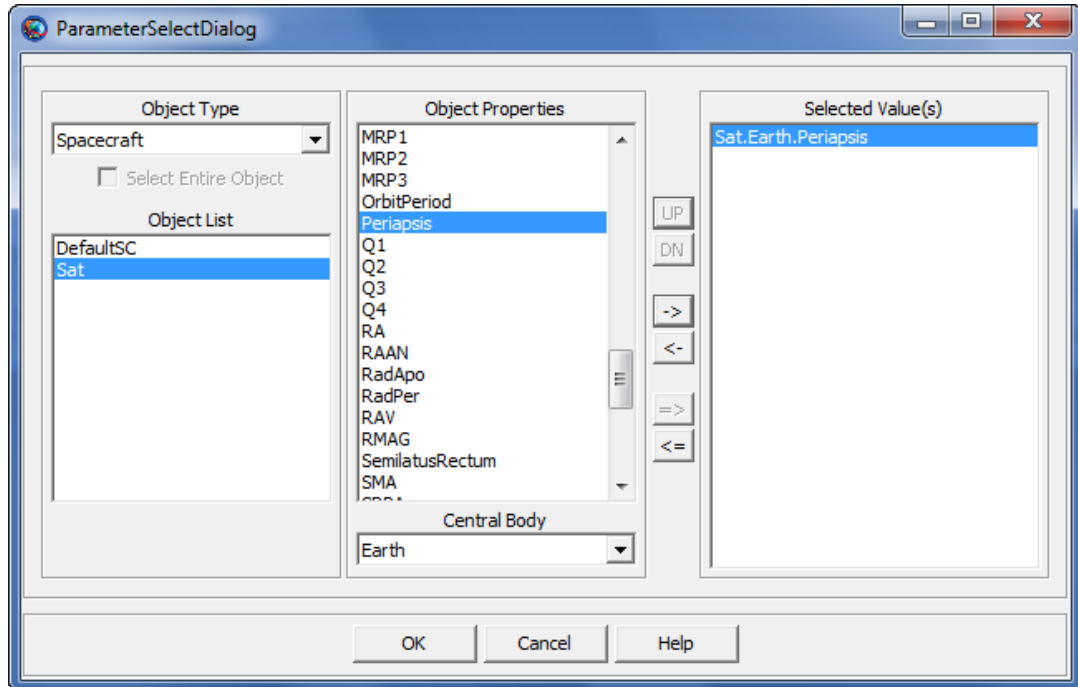


Figure: Propagate Command Parameter Select Dialog Configuration

Having set the propagator, spacecraft, and stopping conditions above, the Propagate1 dialog box should appear as the screen capture below. When you are finished click Ok to save the changes you've made to Propagate1.

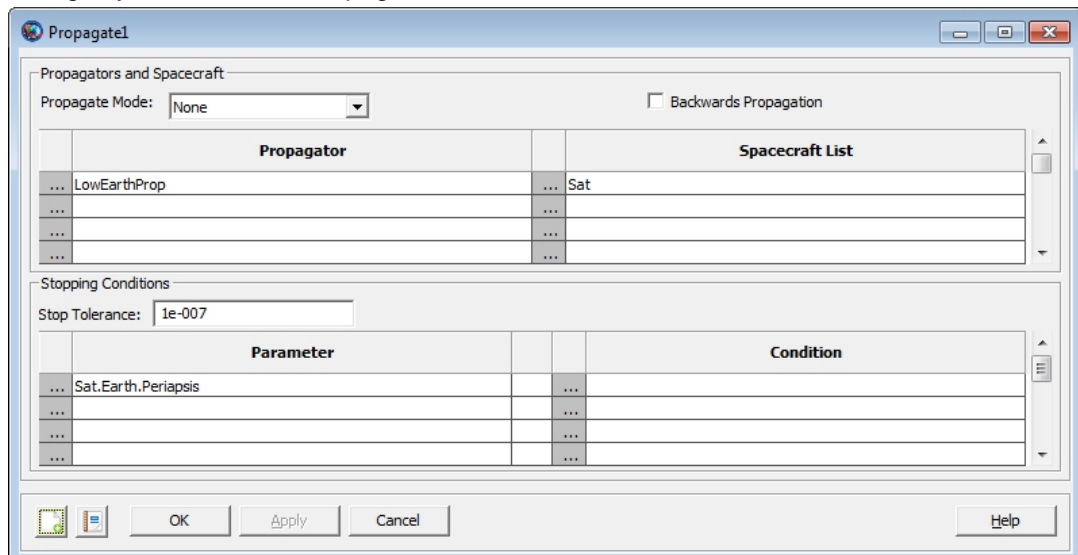


Figure: Propagate Command Configuration

Running the Mission

You have now configured the mission and it is time to run it and see the results.

1. Left-click on the Save button in the toolbar.
2. Left-click on the Run button in the toolbar.

You should see GMAT propagate the orbit and stop at orbit periapsis. The screen capture below summarizes what you should see after correctly completing this tutorial. You can manipulate the Orbit View plot using your mouse to zoom in and rotate the plot.

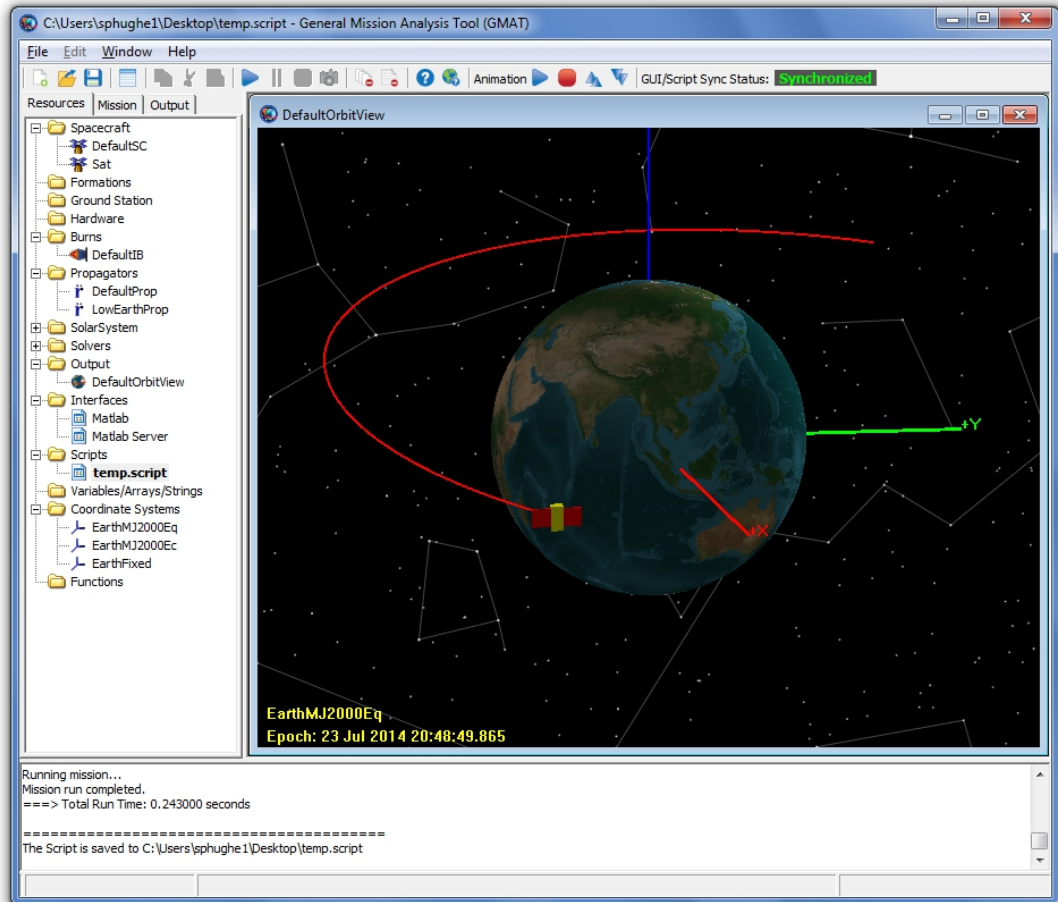


Figure: Orbit View Plot after Mission Run

Designing a Hohmann Transfer

Objective and Overview

Final result: HohmannTransferDesign.script [scripts/HohmannTransferDesign.script]

The objective of this tutorial is to learn how to use differential correction (targeting) to perform a Hohmann transfer from a LEO parking orbit to a GEO mission orbit. In order to focus on the configuration of the differential correction process, you will make extensive use of the default configurations for spacecraft, propagators, and maneuvers. The target sequence employs two velocity-direction maneuvers and two propagation sequences. The purpose of the first maneuver is to raise orbit apogee to 42165 km. The purpose of the second maneuver is to nearly circularize the orbit and yield a final eccentricity of 0.005. The differential corrector and associated commands are used to solve for the required maneuver magnitudes to reach the desired altitude eccentricity. The basic steps of this tutorial are:

1. Create and configure a differential corrector.
2. Modify the default OrbitView to visualize the trajectory.

3. Create two default impulsive maneuvers.
4. Add a target sequence to the mission to raise apogee to GEO altitude and circularize the orbit.
5. Run the mission, save the solution, and rerun the mission using the converged solution.

Creating and Configuring the Resource Tree

Begin by loading the default mission (click the new mission button in the toolbar) or starting a new GMAT session. For this tutorial, we will use the default configurations for a spacecraft (DefaultSC), a propagator (DefaultProp), and maneuvers. DefaultSC is configured to a near circular orbit and DefaultProp is configured to use Earth as the central body with a gravity model of degree and order 4. The default impulsive burn model uses the Velocity Normal Binormal (VNB) coordinate system. You may want to open the dialog boxes for these objects and inspect them more closely as we will leave the settings of those objects at their default values.

Creating the Differential Corrector

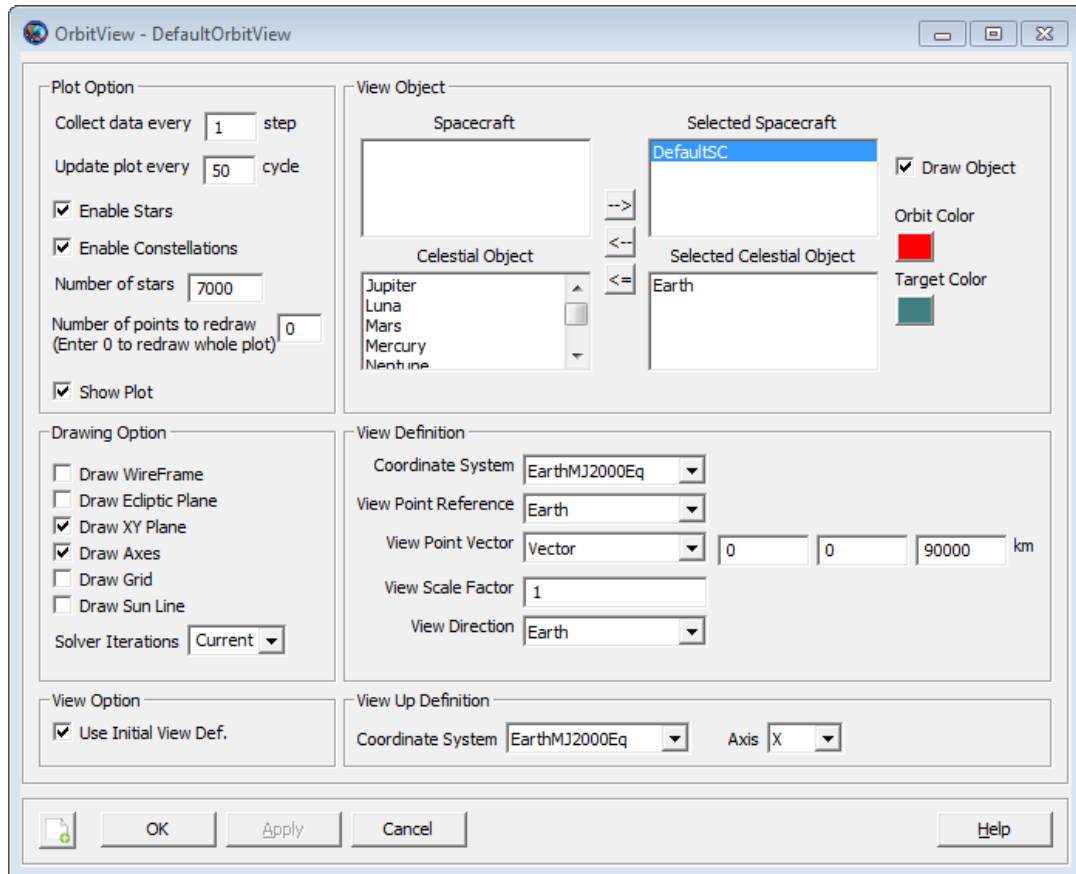
To create a differential corrector:

1. Locate the Solvers folder in the Resource Tree and expand it if it is minimized.
2. Right-click the Boundary Value Solvers folder, select Add, and then select Differential-Corrector.

Modifying the default Orbit View

You need to make minor modifications to default Orbit View so that the entire final orbit will fit in the graphics window.

1. Locate DefaultOrbitView in the Resource Tree, right-click on it, and select Open.
2. Change the SolverIterations input field, located in the Drawing Option group box, to the value Current.
3. Change ViewPointVector to 0, 0, 90000 respectively.
4. Change the ViewUpDefinition Axis to X.
5. Verify the configuration against the screen capture below, make any changes necessary, and click Ok on the DefaultOrbitView dialog box.



Modifications to the Default OrbitView

Creating the Maneuvers.

You need two default maneuvers for this tutorial and we will rename the default maneuver and create a new maneuver:

1. Locate DefaultIB in the Resource Tree, right-click on it, select Rename, and change the name to dv1.
2. Right-click on the Burns folder, select Add-->ImpulsiveBurn.
3. Right-click on the impulsive burn created in the previous step, select Rename, and change the name to dv2.

Creating and Configuring the Mission Tree

Below you will create a targeting sequence to raise orbit apogee to GEOsynchronous altitude (~42165km) and then circularize the orbit. You'll begin by modifying the initial propagate command to propagate to periapsis. Next you will create the command structure and finally you will configure each command.

Initial Propagate Sequence

To configure the initial propagate sequence to propagate to periapsis, perform the following steps:

1. Left-click on the Mission tab to bring up the Mission Tree.

2. Right-click on Propagate1 and select Open from the menu.
3. Locate the Stopping Conditions group box and then right click on the set of ellipses next to the text "DefaultSC.ElapsedSecs". This will open the Parameter Select Dialog box.
4. Under the Object Properties list on the Parameter Select Dialog box, locate periapsis and double-click on it. Click the Ok button to close the Parameter Select Dialog box.
5. Click the Ok button on the Propagate 1 dialog box to save changes and close.

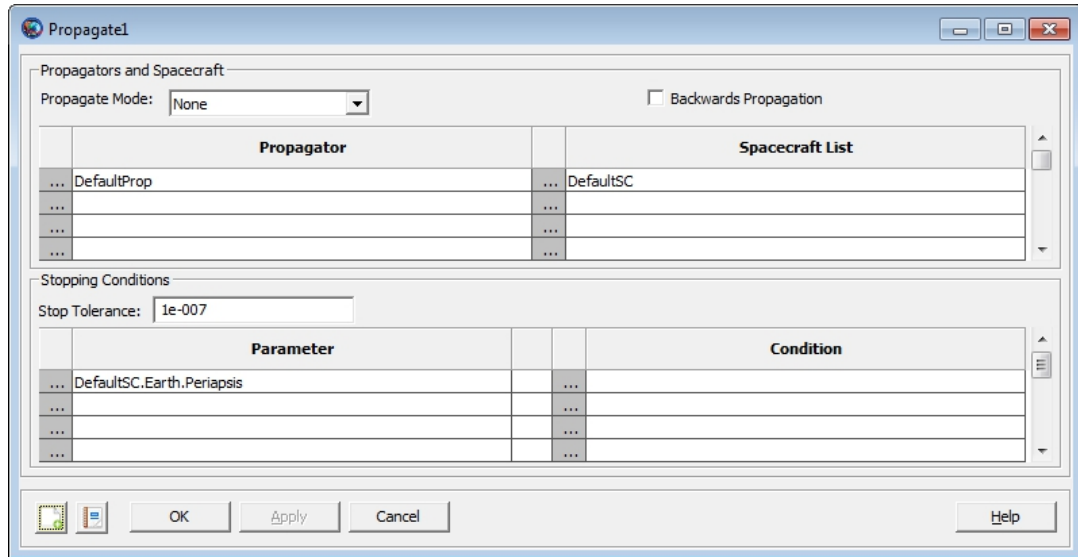


Figure: Propagate1 Command

Creating the Command Sequence

To determine the delta Vs required to raise the orbit apogee to GEO altitude and then circularize the orbit, you will employ a Targeting loop. Let's begin by creating the commands necessary to perform the targeting sequence. The figure below illustrates the configuration of the mission tree after you have completed the steps in this section.

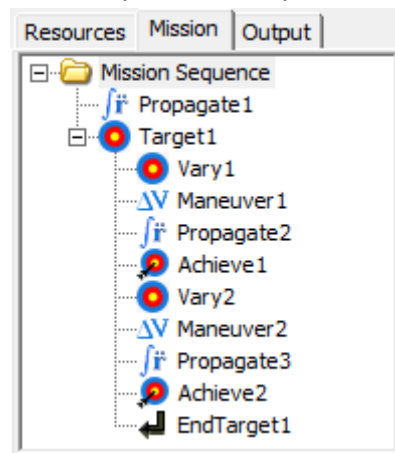


Figure: The Mission Tree for a Hohmann Transfer

1. Right-click on Propagate1 in the Mission Tree, select Insert After, and select Target.
2. Right-click on Target1 in the Mission Tree, select Insert After, and select Vary.
3. Right-click on Vary1 in the Mission Tree, select Insert After, and select Maneuver.

4. Right-click on Maneuver1 in the Mission Tree, select Insert After, and select Propagate.
5. Right-click on Propagate2 in the Mission Tree, select Insert After, and select Achieve.
6. Right-click on Achieve1 in the Mission Tree, select Insert After, and select Vary.
7. Right-click on Vary2 in the Mission Tree, select Insert After, and select Maneuver.
8. Right-click on Maneuver2 in the Mission Tree, select Insert After, and select Propagate.
9. Right-click on Propagate3 in the Mission Tree, select Insert After, and select Achieve.

Let's talk about the function of the command sequence you created above. The Vary commands define the variables the differential corrector can modify to achieve the goals defined in the Achieve commands. Because there are two variables (Vary commands) and two Achieve commands (constraints), this is a "square" targeting problem. Below you will configure the Vary commands to modify the maneuver values to achieve a final orbit radius of 42165 and an eccentricity of 0.005

Configuring the Command Sequence

Now you will configure the commands you created above to solve for the delta-Vs required to perform a Hohmann transfer.

1. Right-click on Target1 in the Mission Tree and select Open. Locate the ExitMode drop-down menu and set it to SaveAndContinue. This will save the converged solution of the targeting problem.
2. When you are finished, click OK to close the Target1 dialog box.

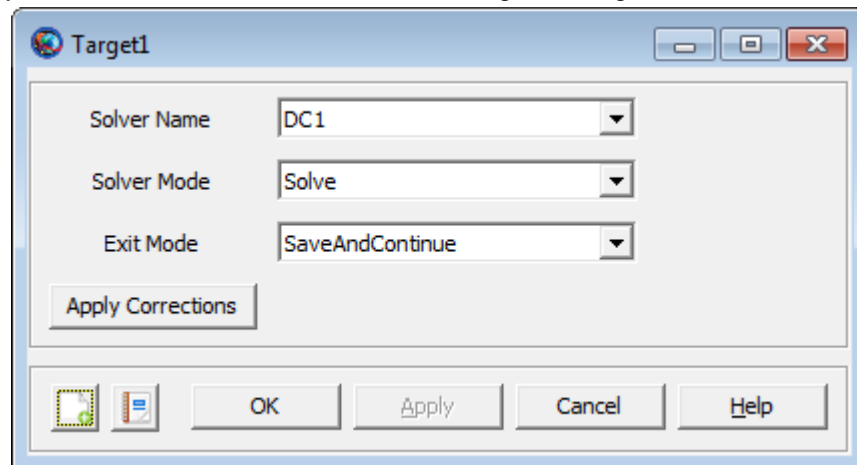


Figure: Target1 Command

3. Right-click on Vary1 in the Mission Tree and select Open. Notice that in the Variable SetUp group box, the variable is defined as dv1.Element1. This is the velocity component of dv1 in the local VNB system. So we do not need to change the targeter variable.
4. Locate the InitialValue text box and set it to 1.0.
5. Set the MaxStep to 0.5 and then click Ok to close the Vary1 dialog box.

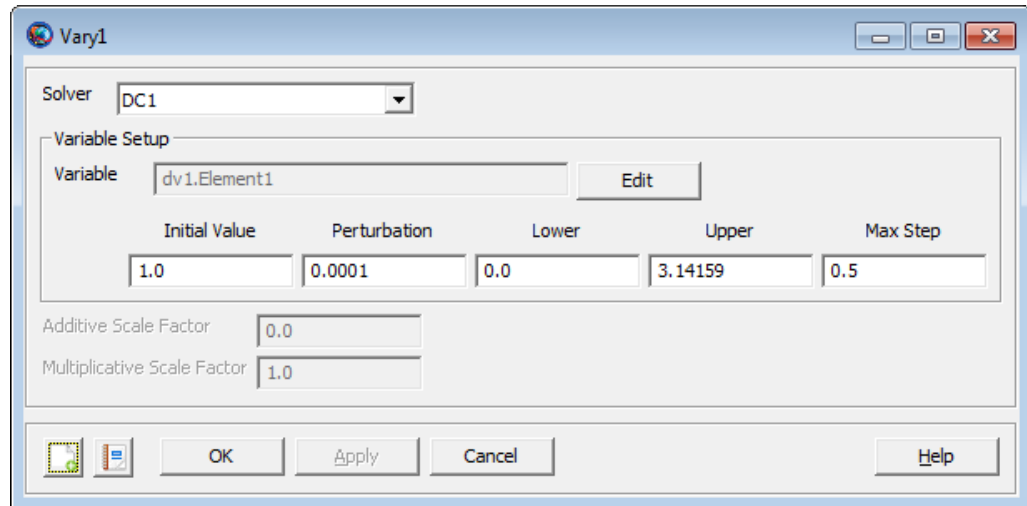


Figure: Vary1 Command

6. Double-click on the Maneuver1 command in the Mission Tree. Notice that the command is set to apply dv1 to DefaultSC so we do not need to change any settings for this command. Click Ok to close the Maneuver1 dialog box.

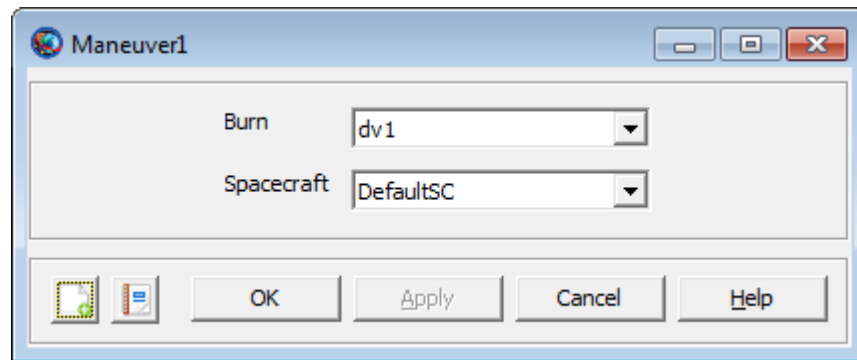


Figure: Maneuver1 Command

7. Double-click on the Propagate2 dialog box. Use the same procedure shown in the sections above to set the stopping condition to Apoapsis and then click Ok to close the dialog box.

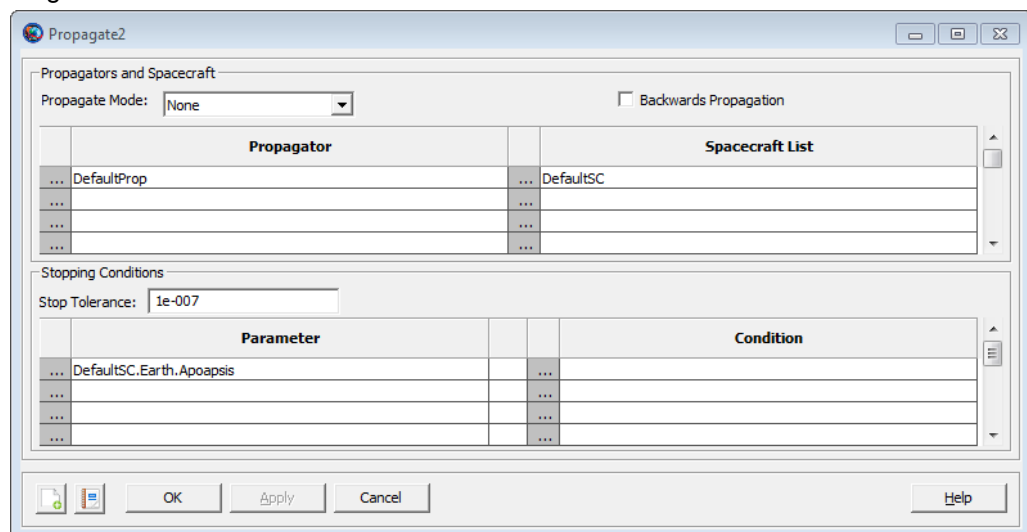


Figure: Propagate2 Command

8. Double-click on the *Achieve1* command in the Mission Tree. Notice that the goal is set to DefaultSC.Earth.RMAG. So we do not need to change any settings for this command. Click Ok to close the dialog box.

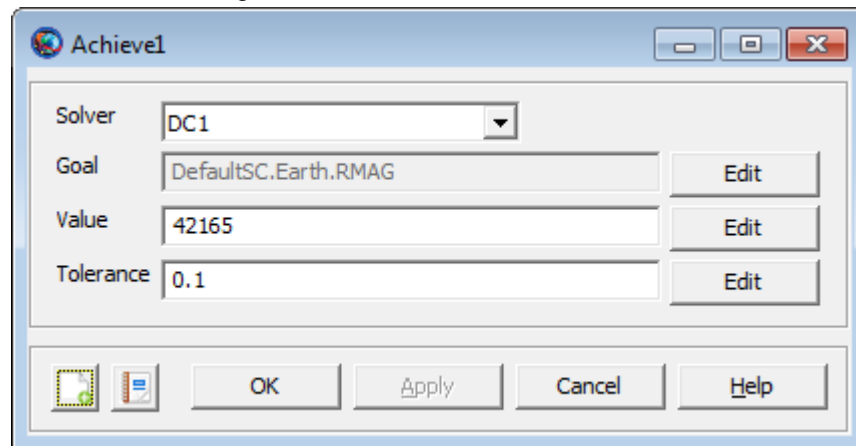


Figure: Achieve1 Command

9. Double-click on Vary2 in the Mission Tree.
10. Click the Edit button next to Variable text box.
11. Click on dv2 under the Object List.
12. Double-click on Element1 under the ObjectProperties.
13. Click Ok to close the ParameterSelect dialog box.

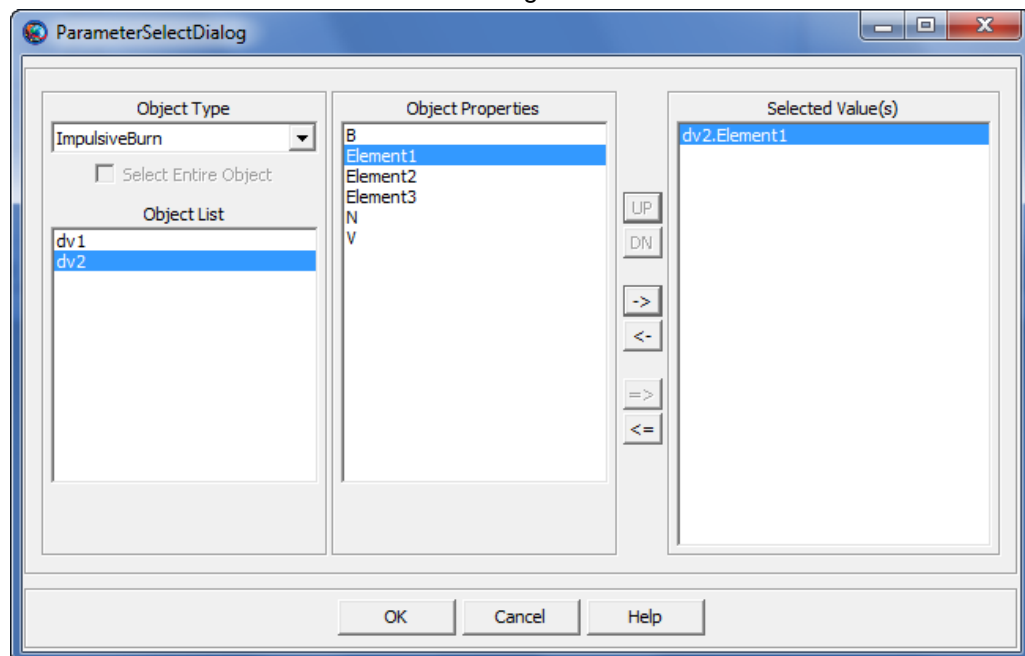


Figure: Parameter Select for Vary2

14. Locate the initial value text box and set it to 1.0.
15. Locate the MaxStep text box and set it to 0.2. Click Ok to close the Vary1 dialog box.

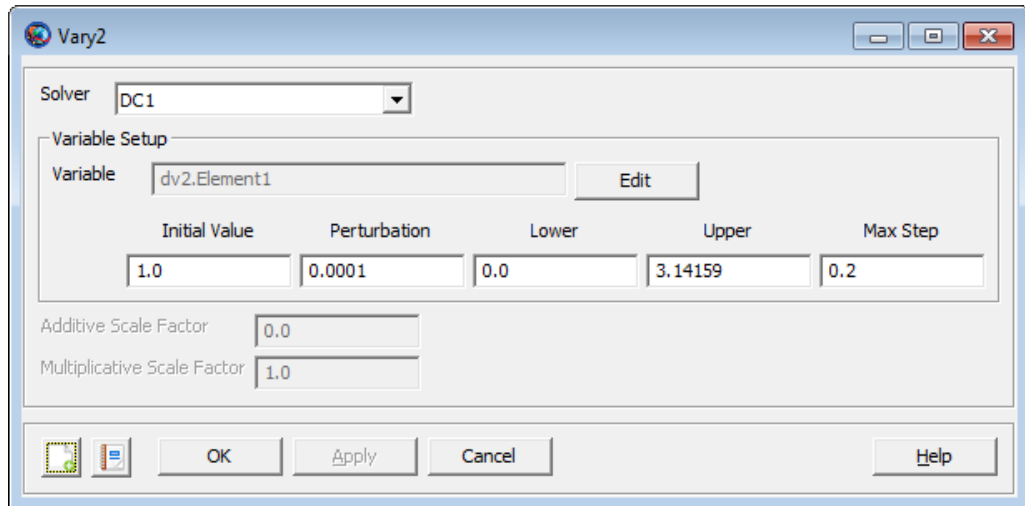


Figure: Vary2 Command

16. Double-click on the Maneuver2 command in the Mission Tree.
17. Locate the Burn combo box and change it to dv2. Click OK to close the Maneuver2 dialog box.

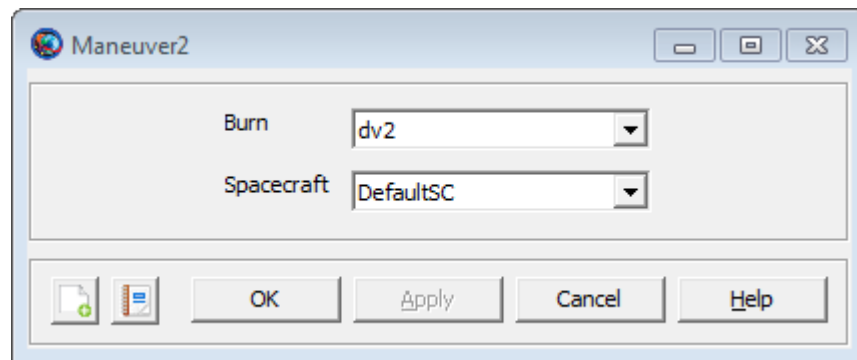


Figure: Maneuver2 Command

18. Double-click on the Propagate3 dialog box. Use the same procedure shown above to set the stopping condition to ElapsedDays of 1.0. box.

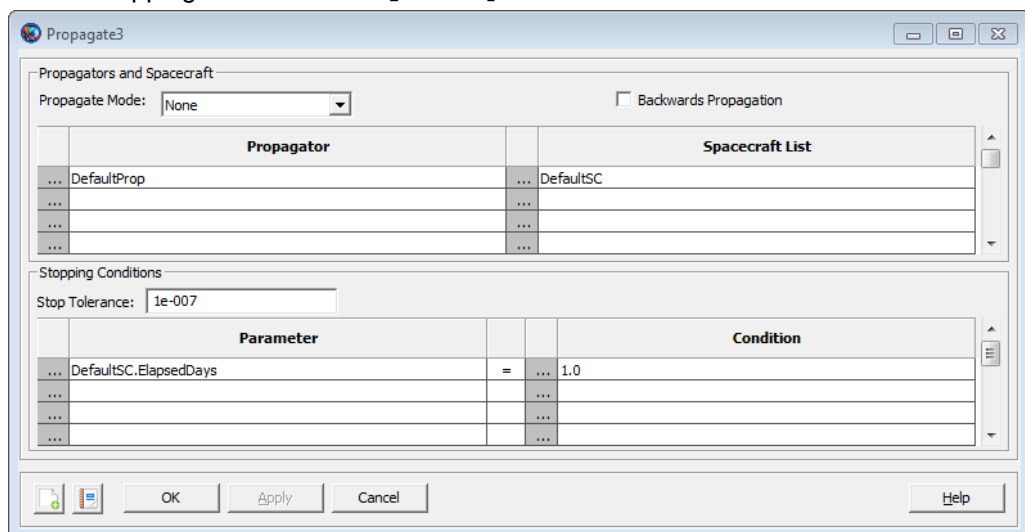


Figure: Propagate3 Command

19. Finally, double-click on the Achieve2 command in the Mission Tree.
20. Click on the Edit button next to the Goal text box.
21. Locate `ECC` in the Object Properties list and double-click on it. Click Ok to close the Parameter Select Dialog Box.
22. Change the `Value` to 0.005.
23. Change the `Tolerance` to 0.0001

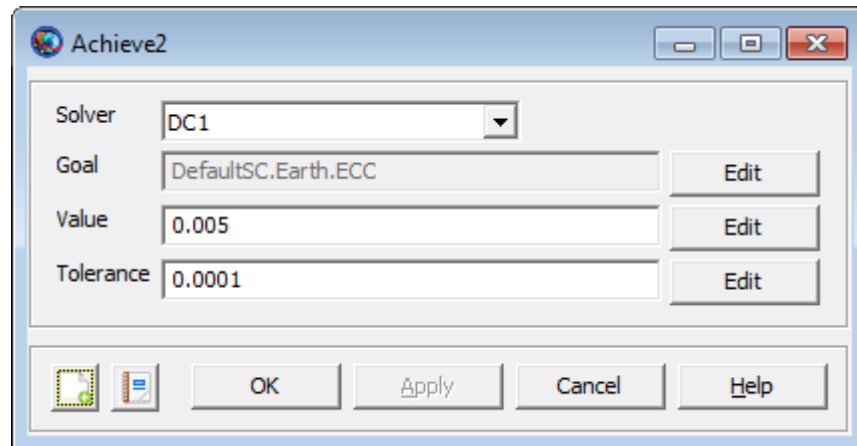


Figure: Achieve2 Command

Running the Mission

Before running the mission, click on the Save button in the toolbar and save your file to the desired location. Now click on the Run button in the Toolbar. As the mission is running, you will see the Differential Corrector try different maneuvers values resulting in trajectory iterations in light blue. These are Differential Corrector iterations, and perturbation passes to determine the Jacobian of the two-point boundary value problem. After the mission is run, the OrbitView should appear similar to the image shown below.

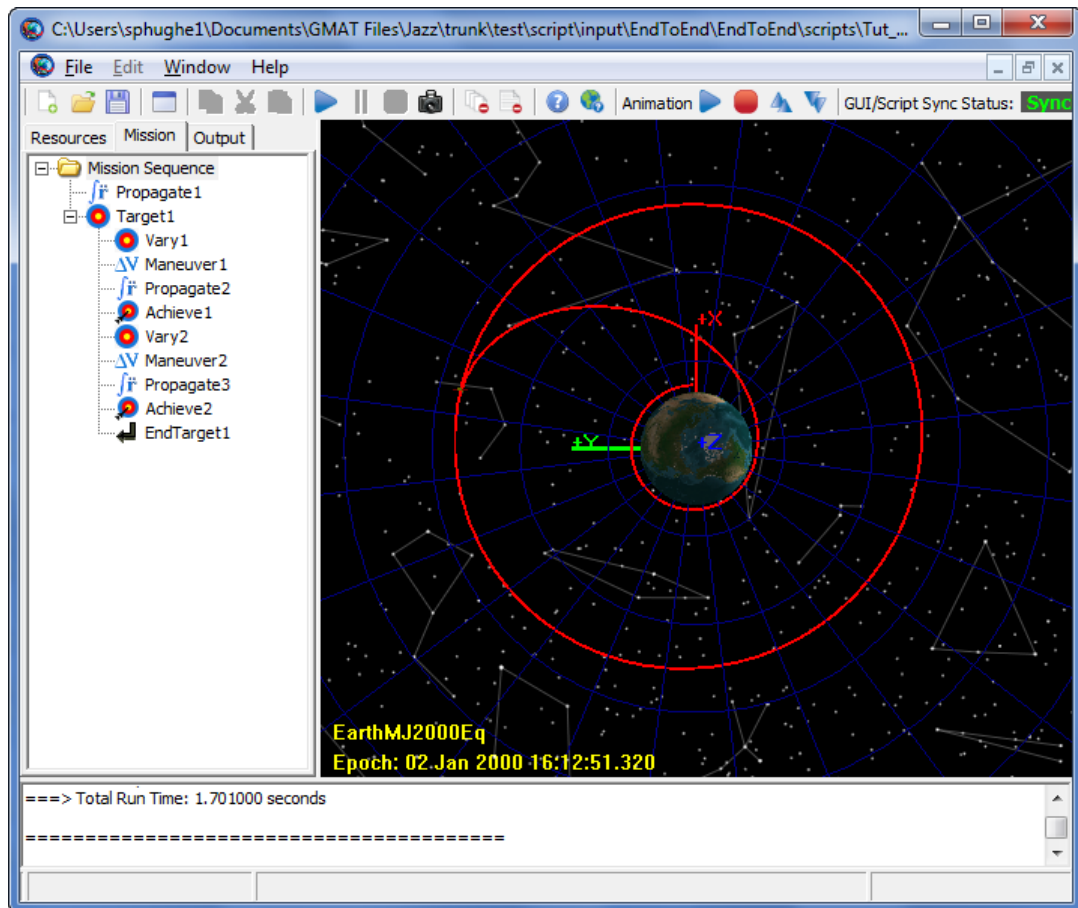


Figure: Output After Final Propagate Sequence

You can save the resulting solution so that if you make small changes to the problem and retarget, the initial guess for subsequent runs will use the solution from your work above. .

1. Double-click on Target1 in the Mission Tree.
2. Left-click on Apply Corrections.
3. Rerun the mission by clicking the Run button in the toolbar. If you inspect the results in the message window you should see that the targetting only took one iteration because it started from the solution!

LEO Station Keeping

Objective and Overview

Final result: HohmannTransferDesign.script [scripts/HohmannTransferDesign.script]

In this tutorial, you will learn how to use control flow and targeting in GMAT to maintain a Low-Earth orbit in the presense of drag. The area-to-mass ratio of the spacecraft is large to cause a rapid lowering of the orbit semimajor axis for the sake of simulation time. However, the process used in this script is useful for generating delta-V estimates for LEO stationkeeping of real-world missions. The basic steps of this tutorial are:

1. Create and configure a spacecraft, impulsive maneuver, propagator, XYPlot, and differential corrector.
2. Create a conditional loop using a while statement that propagates for 10 days.

3. Run the mission and observe the behavior if there is no orbit control strategy.
4. Create a target sequence nested in an if statement that executes if altitude is below 342 km.
5. Run the mission and observe the behavior of orbit altitude with the control strategy implemented in step 4.

Creating and Configuring the Resource Tree

In this section, you will configure a model of a LEO spacecraft, a propagator, a maneuver, and an XY plot to visualize the SMA during the control sequence developed in the next section.

Creating the Spacecraft

1. Starting from the Resource Tree, right-click on DefaultSC, select rename, and change the name to LEOsat.
2. Right-click on LEOsat, locate the EpochFormat combo box, and change the Epoch Format to UTCGregorian.
3. Set the Epoch to 05 Jan 01 2010 00:00:00.000
4. Set the state using the following values: X = -4083.9 , Y = 4691.8 , Z = -2576.7 , VX = -4.252 , VY = -5.509 , VZ = -3.284
5. Verify the configuration against the screen capture below, make any changes necessary, and click Ok on the LEOsat dialog box.

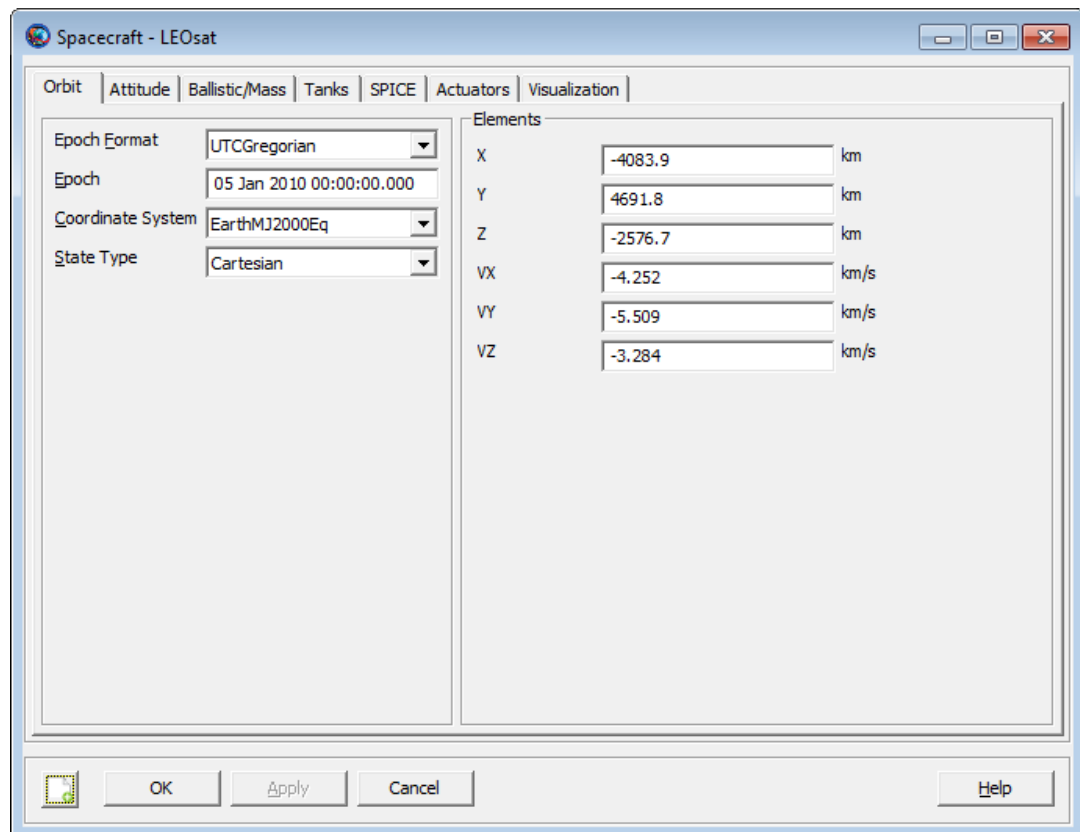


Figure: LEOsat Spacecraft Object

Creating the Propagator

1. Right-click on DefaultProp, select Rename, and change the name to LEOprop.
2. Right-click on DefaultProp, and select Open.
3. Locate the Drag Model group box and change the Atmosphere model to JacchiaRoberts.
4. Click the select button to the right of the Point Masses list. Add the Sun, and Luna as Point Masses to the right-hand side list and click Ok.
5. Check the Use Solar Radiation Pressure box
6. Verify the configuration against the screen capture below, make any changes necessary, and click Ok on the LEOprop dialog box.

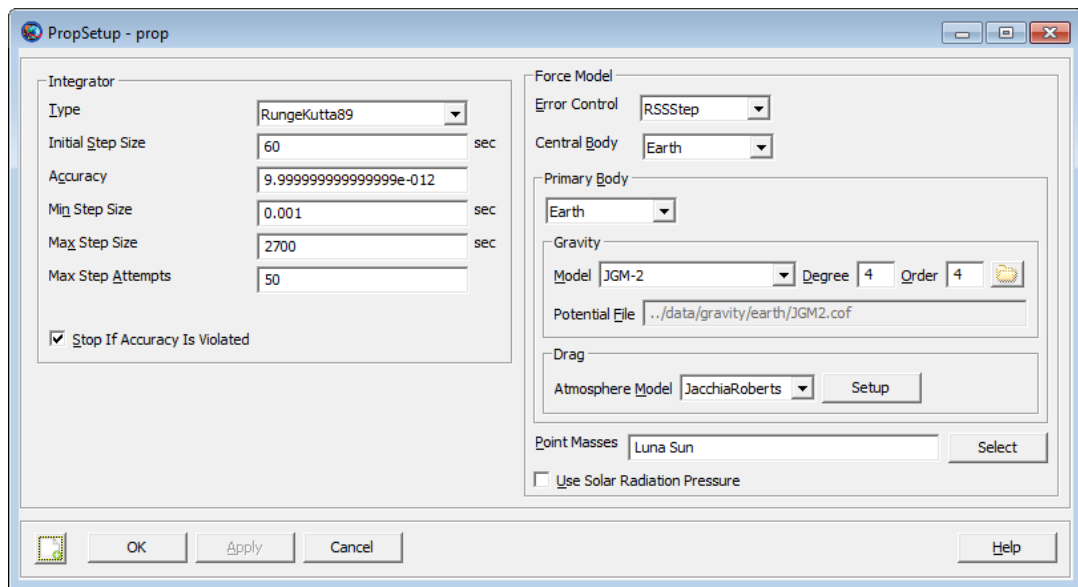


Figure: LEOprop Propagator Object

Creating the DifferentialCorrector and Maneuver

We'll use the default configurations for a maneuver and a differential corrector so simply follow the steps below to create default configurations of those objects.

1. Locate the Solvers folder under the Resource Tree and expand it if it is minimized.
2. Right-click on the Boundary Value Solvers folder and select DifferentialCorrector.
3. Rename the new Differential Corrector DC using steps described above.
4. Right-click on DefaultIB in the Burns folder, select Rename, and change the name to DeltaV.

Creating and Configuring the Mission Sequence

You will implement this mission in two parts. The first part is to add a While loop to the mission sequence, put a propagate sequence in the While loop, and configure the loop to propagate for about 10 days. At this point in the tutorial, there is no orbit control strategy implemented and upon running the mission you will see the SMA drop below the allowed value of 340 km. In the second part, you will implement a control sequence to ensure SMA remains above approximately 340 km. Below is an illustration of the Mission Tree after you have completed Part 1.

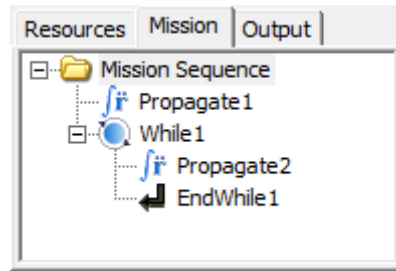


Figure: Mission Tree, Part I

1. Starting from the Mission Tree, right-click on Propagate1, select Insert After, locate Control Logic in the menu, and select While.
2. Right-click on Propagate1 and select Delete.
3. Right-click on While1, select Insert After, and select Propagate.
4. After While1, insert another Propagate

Now you will configure the While1 to continue if the ElapsedDays of LEOsat is less than 10 days and configure the Propagate2 command to take a single propagation step.

1. Open the dialog box for the Propagate1 command.
2. Double-left click in the Stopping Conditions *Parameter* field to select the text `LEOSat.ElapsedSecs`. Press the delete key to delete the text.
3. Double-left click in the Stopping Conditions *Condition* field to select the text `12000`. Press the delete key to delete the text.
4. Click OK to close the Propagate1 dialog box.

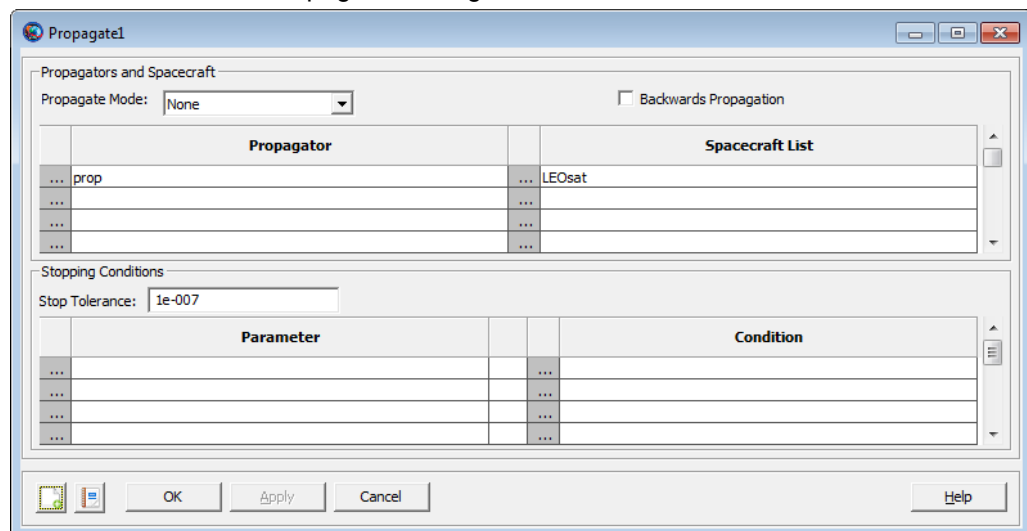


Figure: Propagate1 Command

5. Open the dialog box for the While1 command.
6. Notice that the left-hand side of the conditional statement is already set to `LEOSat.ElapsedDays` and the conditional operator is already set to "less-than". We will leave those fields at their default.
7. Change the right-hand side text box to the value `10.0`.
8. Click OK to close the While1 dialog box.



Figure: While1 Command

Now let's run the mission and observe the resulting altitude evolution by clicking Run in the Toolbar. You should see a plot that looks similar to the illustration below. Notice that the altitude drops below the allowed value of 342 km. In the next part of this tutorial, you will implement a control strategy to prohibit altitude from breaking the constraint.

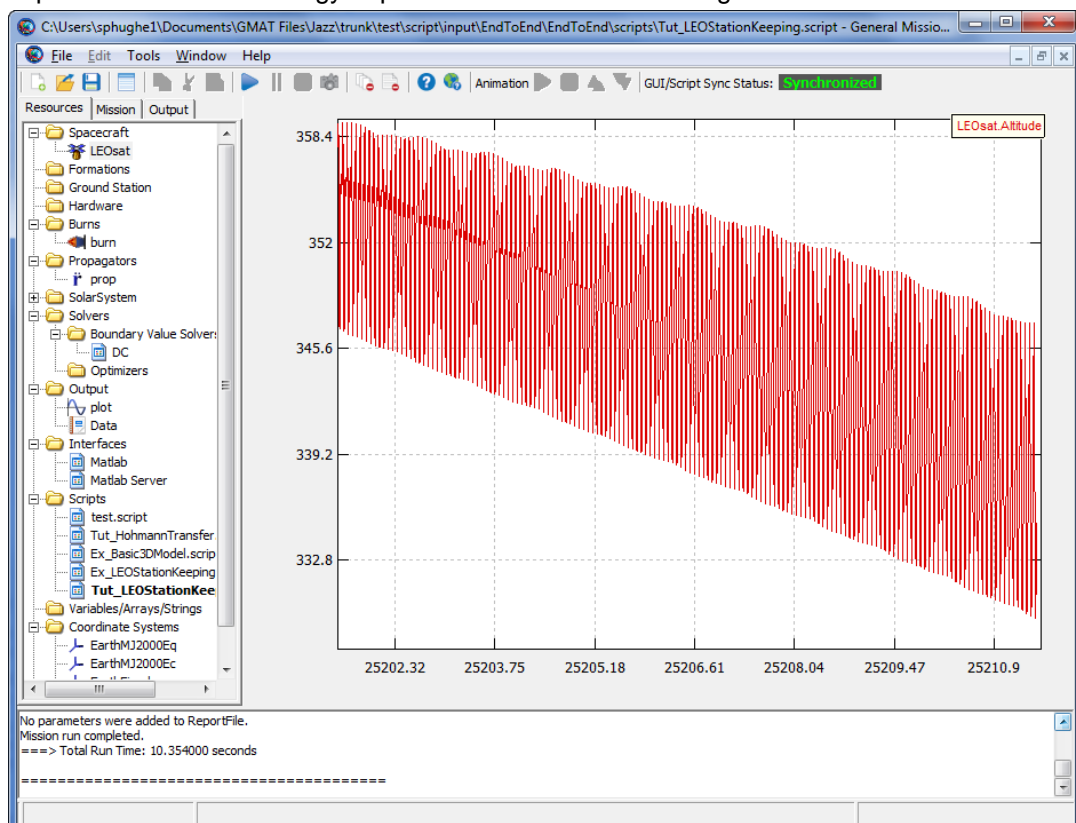


Figure: Output After Part 1

Below you will implement a control strategy that performs a Hohmann transfer if altitude drops below 342 km. If you have not taken the Hohmann Transfer tutorial you may wish to do so now. Below we assume you have basic knowledge of how to configure a Hohmann transfer

in GMAT and do not provide all detailed steps. The figure below illustrates the Mission Tree after you have completed configuring the command sequence below.

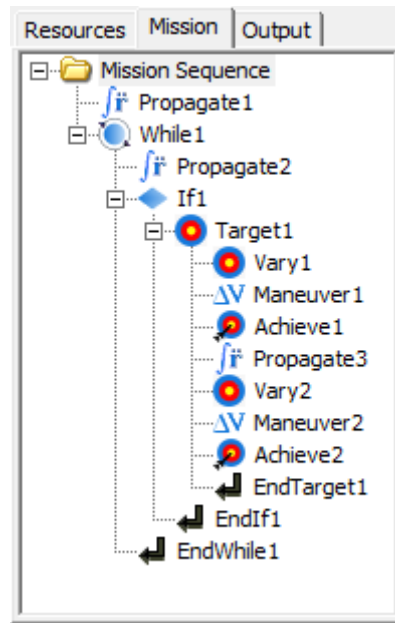


Figure: Mission Tree, Part II

1. Add an If statement after the Propagate2 Command.
2. Add a Vary statement inside of the Target1 Command.
3. Add a Maneuver after the Vary1 Command.
4. Add an Achieve command after the Vary1 Command.
5. Add a Propagate command after the Achieve1 Command.
6. Add a Vary Command after the Propagate3 Command.
7. Add a Maneuver after the Vary2 Command.
8. Add an Achieve after the Maneuver2 Command.

Now you will configure the commands you created above to perform the Hohman transfer control sequence.

1. Open the dialog box for the If1 command
2. Set the left-hand side value to LEOSat.Altitude
3. Set the right-hand side value to 342 and click OK to close the If1 dialog box.

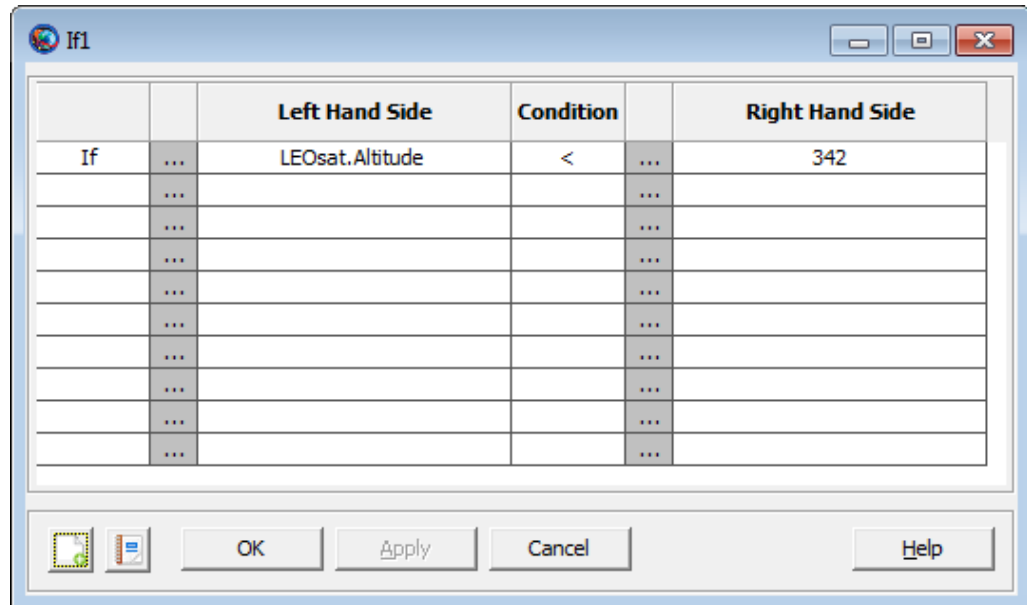


Figure: If1 Command

4. Open the dialog box for the Vary1 Command.
5. Change the IntialValue field to 0.002 and click OK to close.

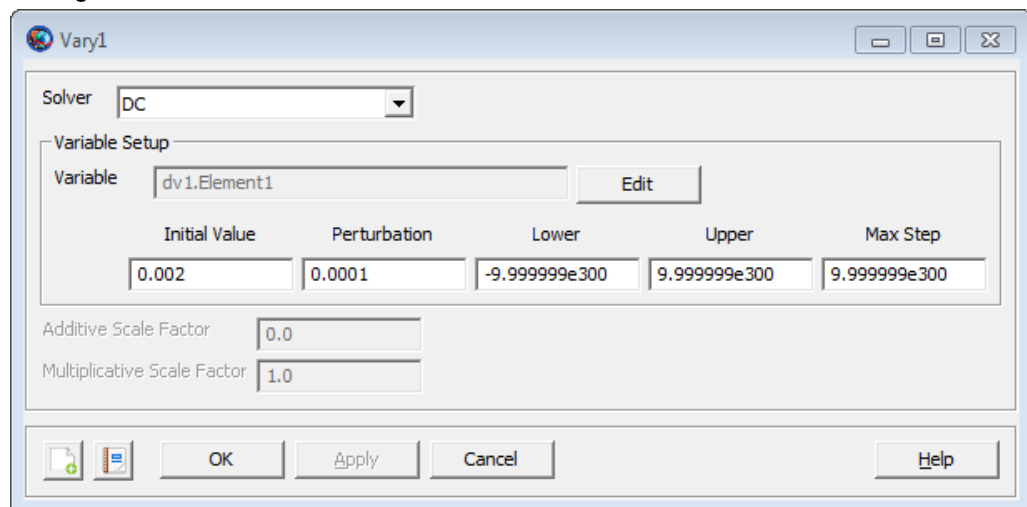


Figure: Vary1 Command

6. Open the dialog box for the Achieve1 Command.
7. Change the left hand side value to LEOsat.Earth.SMA.
8. Change the right hand side to 6734 and then click OK to close.

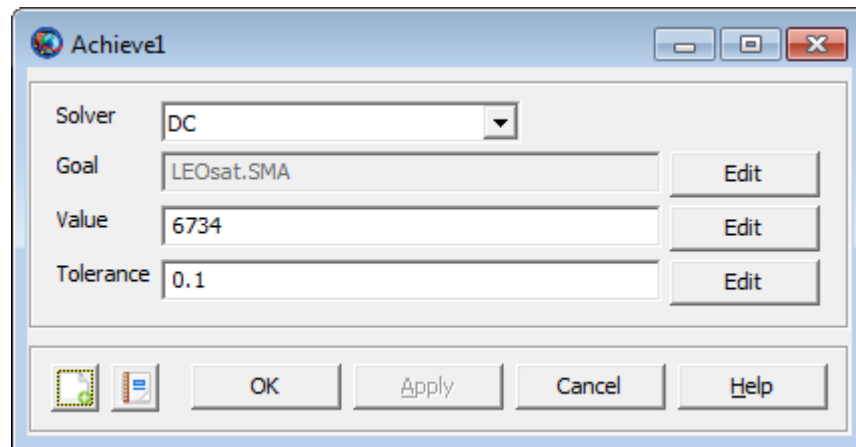


Figure: Achieve1 Command

Running the Mission

The views that you created are good for seeing how well the mission sequence is keeping LEOsat in its orbit. The SMA view shows some thickness to it while the RAAN view shows barely any at all. You may now try and change the mission parameters as outlined in the mission tree section to see how the spacecraft's drift is affected.

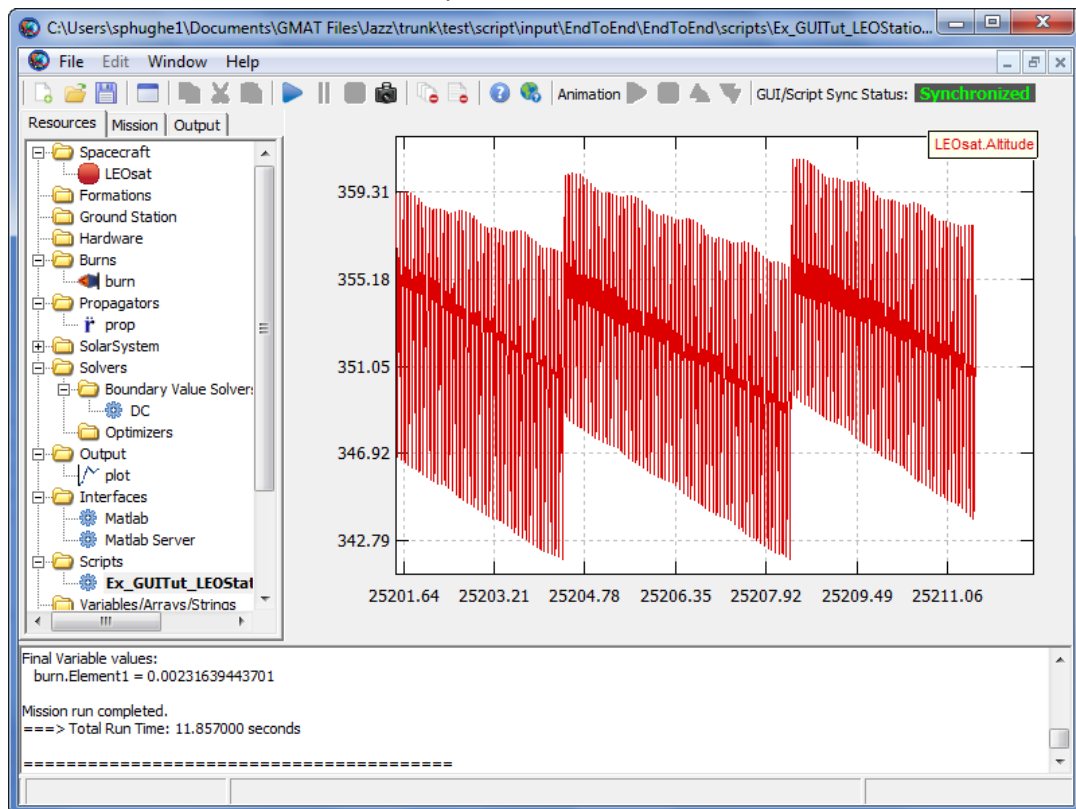


Figure: Output

Algebraic Optimization

Objective and Overview

This tutorial finds the minimum value to satisfy a function. This tutorial is intended to show how GMAT's optimizer works. Uses of optimization in a true mission include minimizing the amount of fuel or minimum flight time required to achieve certain characteristics. Learning how to optimize a mission sequence also involves learning about optimizers, nonlinear constraints, and the minimize command.

You can download the script file and run it beforehand to see the final results of this tutorial: `Ex_AlgebraicOptimization.script` [scripts/Ex_AlgebraicOptimization.script]

Prerequisites

- Basic Understanding of how to create and propagate a spacecraft, as in Tutorial Creating and Propagating a Spacecraft

Mission Description

- Objective: The goal here is to find values of variables X1 and X2 that minimize a criterion function F, with a constraint that $X1+X2=8$, as follows:

$$F = (X1 - 2)^2 + (X2 - 2)^2$$

$$G = X1 + X2$$

$$G = 8$$

- Find: Values X1 and X2

Creating and Configuring the Resource Tree

Objects Required

- Optimizer: SQPfmincon
- Plots/Reports: Report Data
- Variables: X1, X2, F, G

Creating and Modifying Objects

- Variables/Strings/Arrays
 - Variable X1 = 0
 - Variable X2 = 0
 - Variable F = 0
 - Variable G = 0
- Solvers
 - SQPfmincon Optimizer (Figure below shows the resulting SQPfmincon window)

- Tol Fun = .0001
- Tol Con = .0001
- Tol X = .01
- Max Fun Evals = 1000
- Max Iter = 250
- Diff Min Change = .000001
- Diff Max Change = .01
- Display : Iter
- Diagnostics

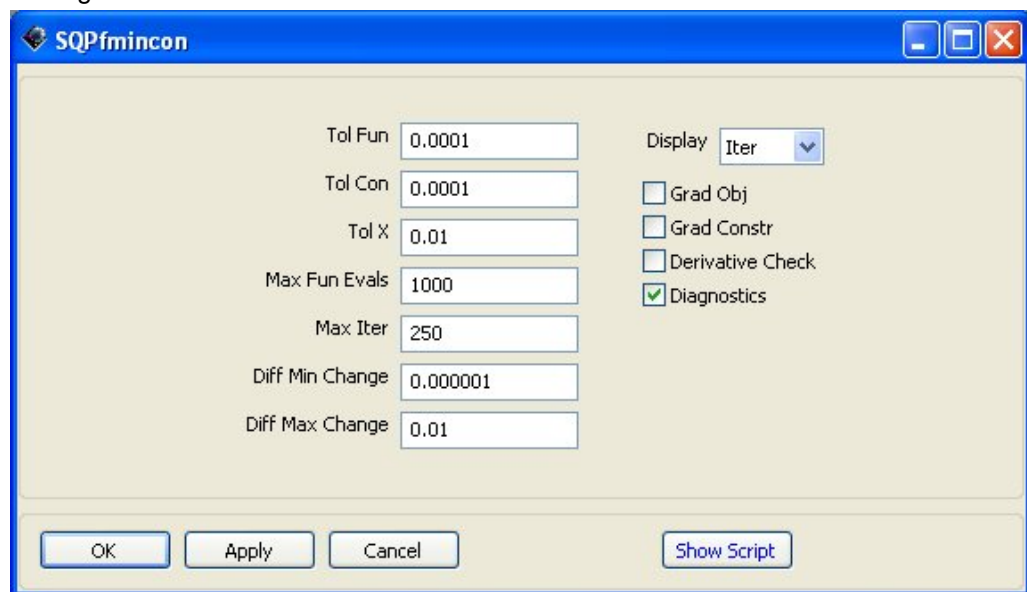


Figure: SQPfmincon Boundary Value Solver Object

- Plots/Reports
 - Data ReportFile
 - Clear Parameter List
 - Saved at ./output/SampleMissions/Ex_AlgebraicOptimization.report

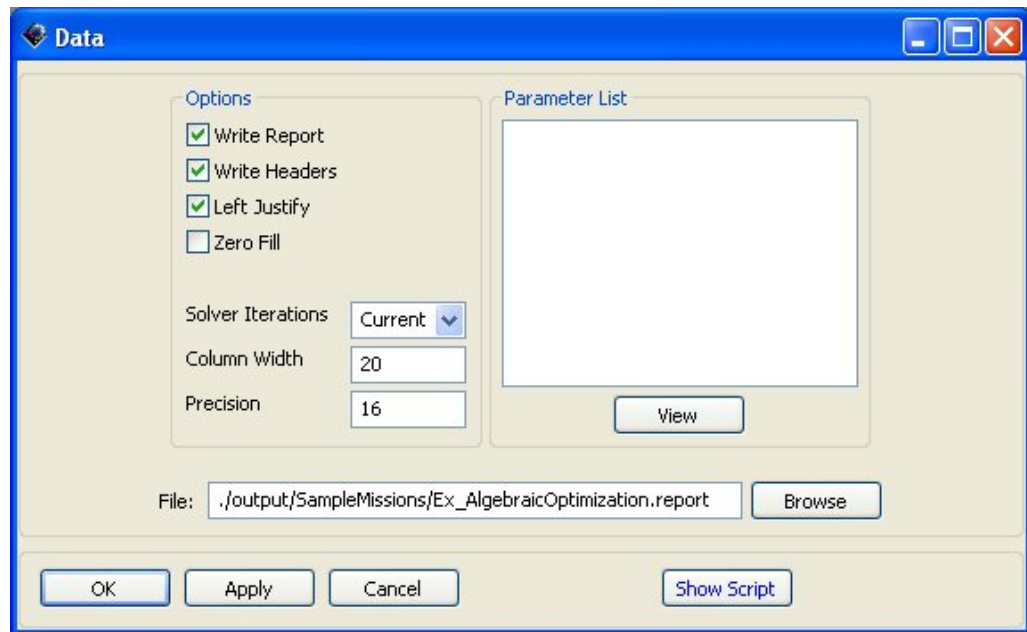


Figure: Data ReportFile System Object

Creating and Configuring the Mission Tree

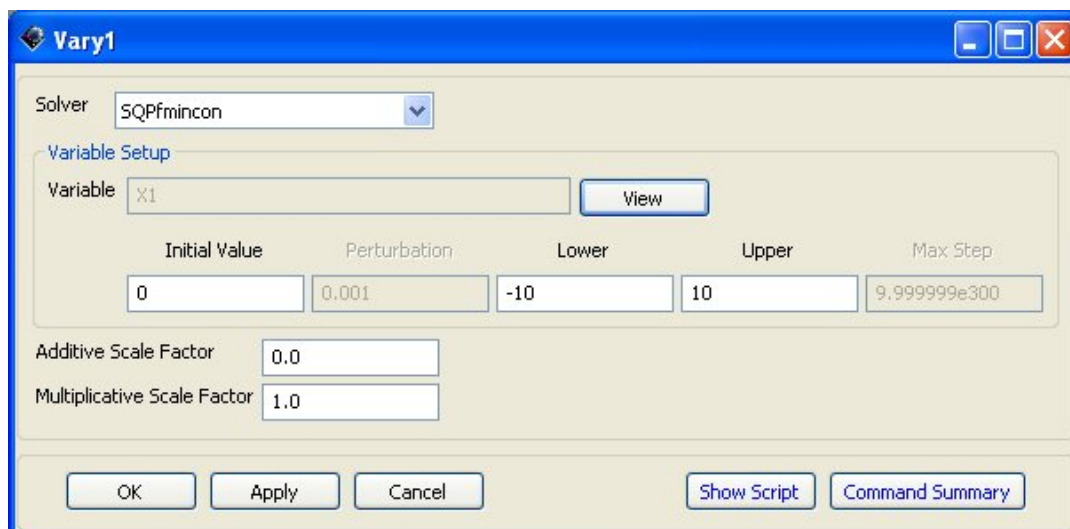
Resource Mission and Output Trees

As said above, values X_1 and X_2 are variables of function F that is to be optimized. The sum of these values form a constraint G , $G=X_1+X_2=8$. The final values of F , X_1 , X_2 are sent to a report file.

Creating and Modifying Command Sequences

Set up the command sequence as in the Optimize sequence below. Optimization is done in Equation1, via the optimization criterion function F , using the Matlab routine SQPfmincon. For the constraint G , since a function cannot be a constraint in itself, the value G is used. This is done in Equation2.

- Optimize
 - Vary X_1 using SQPfmincon
 - Upper = 10, Lower = -10
 - Vary X_2 using SQPfmincon
 - Upper = 10, Lower = -10
 - Equation1 $F = (X_1 - 2)^2 + (X_2 - 2)^2$
 - Minimize the value of F using SQPfmincon
 - Equation2 $G = X_2 + X_1$
 - Nonlinear Constraint $G = 8$ using SQPfmincon
- Report to file Data the values of variables F , G , X_1 , X_2



Vary1

Solver: SQPfmicon

Variable Setup

Variable: X1 [View](#)

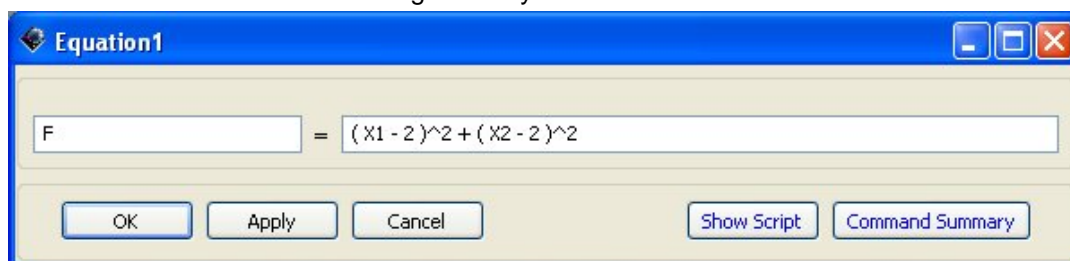
Initial Value	Perturbation	Lower	Upper	Max Step
0	0.001	-10	10	9.999999e300

Additive Scale Factor: 0.0

Multiplicative Scale Factor: 1.0

[OK](#) [Apply](#) [Cancel](#) [Show Script](#) [Command Summary](#)

Figure: Vary X1 Command

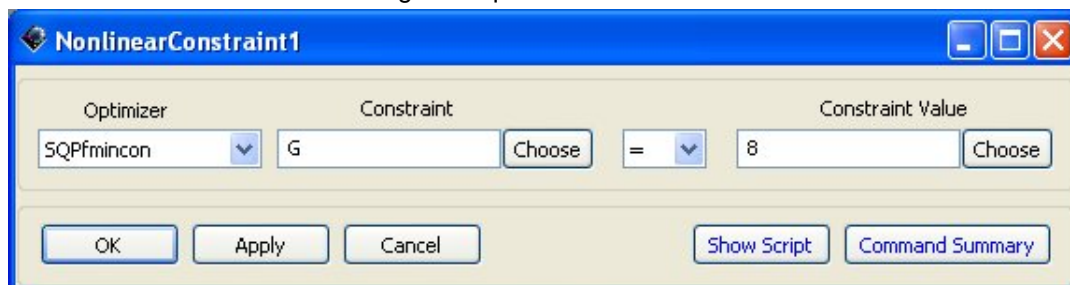


Equation1

F = $(X1 - 2)^2 + (X2 - 2)^2$

[OK](#) [Apply](#) [Cancel](#) [Show Script](#) [Command Summary](#)

Figure: Equation1 Command



NonlinearConstraint1

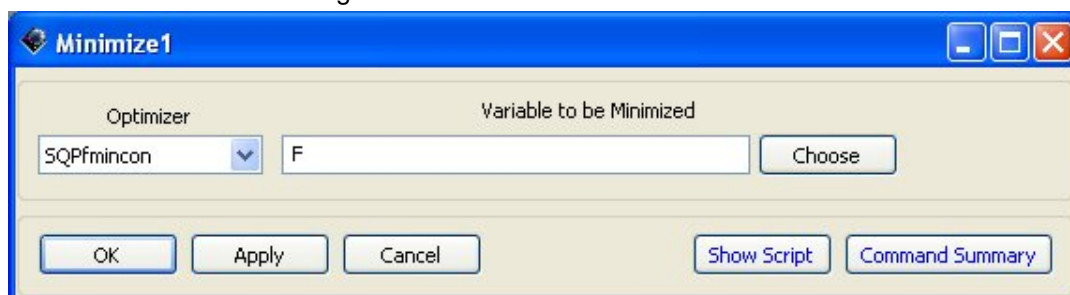
Optimizer: SQPfmicon

Constraint: G [Choose](#)

= [=](#) Constraint Value: 8 [Choose](#)

[OK](#) [Apply](#) [Cancel](#) [Show Script](#) [Command Summary](#)

Figure: Nonlinear Constraint Command



Minimize1

Optimizer: SQPfmicon

Variable to be Minimized: F [Choose](#)

[OK](#) [Apply](#) [Cancel](#) [Show Script](#) [Command Summary](#)

Figure: Minimize F Command

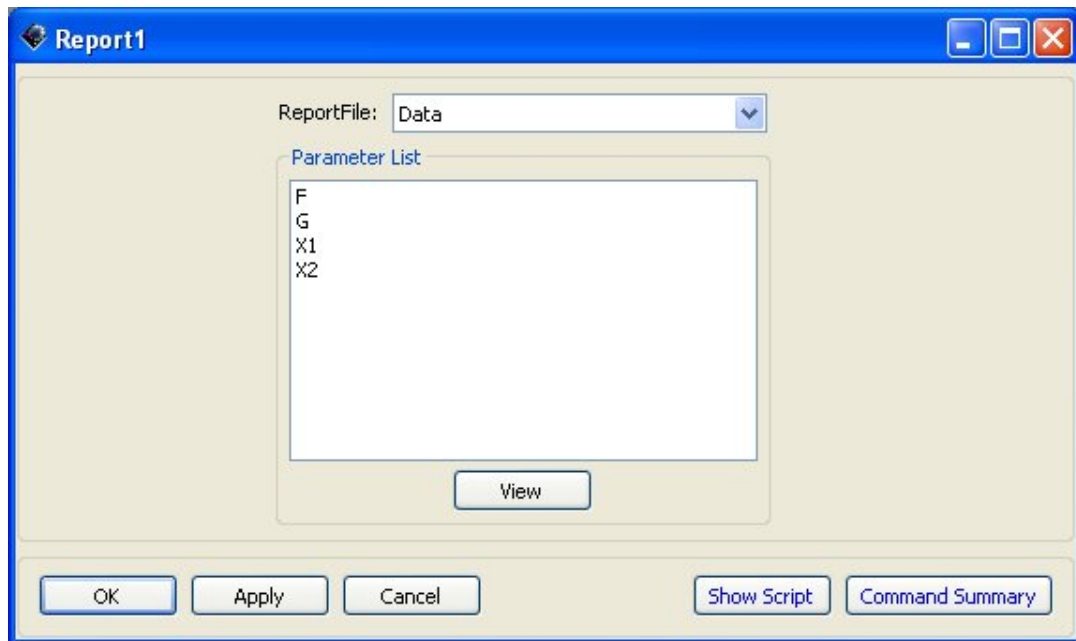


Figure: Report Command

The resulting view of Resource, Mission, and Output Trees

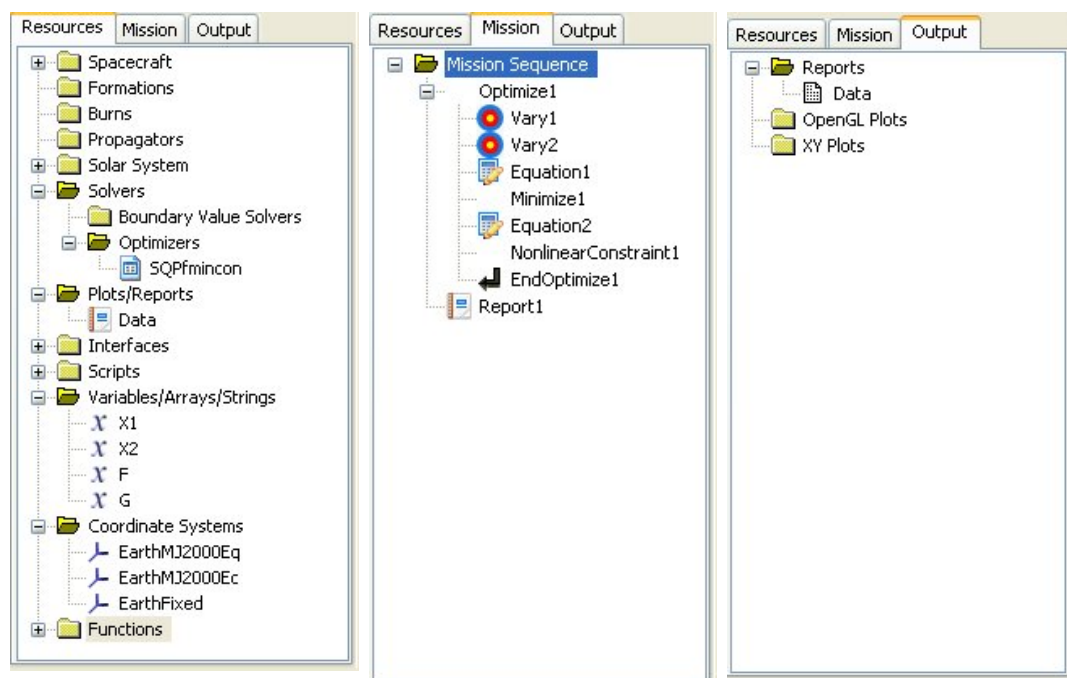
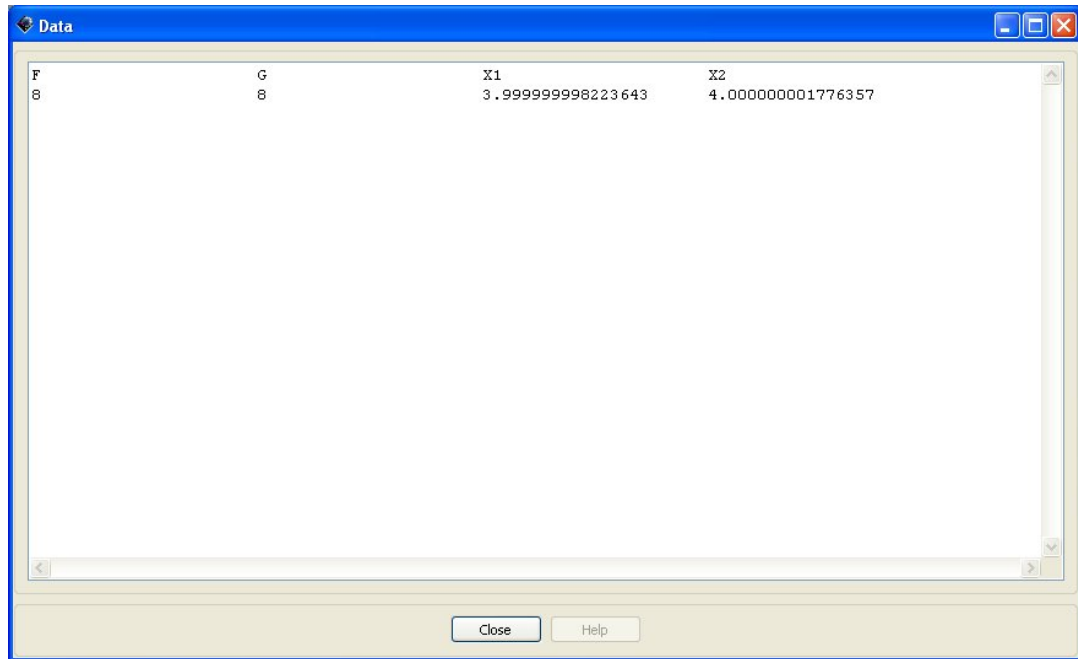


Figure: Algebraic Optimization Resource, Mission, and Output Trees

Running the Mission

There is no plot for this mission sequence. Its results - values X1, X2, and the minimum value of the optimized function F - can be found by calling ReportFile Data.



F	G	X1	X2
8	8	3.999999998223643	4.000000001776357

Figure: Data ReportFile

Mars B-Plane Targeting

Objective and Overview

The goal of this mission is to show how to target a transfer trajectory from an Earth parking orbit to a Mars orbit via B-Plane targeting. Given a specified inclination for the final orbit, we solve for the velocity of the spacecraft in certain directions in order to determine a transfer trajectory.

You can download the script file and run it beforehand to see the final results of this tutorial: MarsBPlaneTutorial.script [scripts/MarsBPlaneTutorial.script]

Prerequisites

- Basic understanding of B-Planes and their usage in targeting a position.
- Tutorials:
 - Creating and Propagating a Spacecraft
 - Designing a Hohmann Transfer

Mission Description

- Objective: Target a Martian transfer trajectory using B-Plane Targeting, to achieve a final inclination of 24.677° .
- Assume:
 - Parking orbit SMA = 10000 km
 - Parking orbit ECC = 0.1
 - INC = 25.0°

- $RAAN = 332.5^\circ$
- $AOP = 90.0^\circ$
- $TA = 360^\circ$
- Find:
 - INC
 - BdotT
 - BdotR

Creating and Configuring the Resource Tree

For this tutorial, four impulsive burns are used to maneuver a single spacecraft, and three propagators are used for different portions of the spacecraft's propagation. The propagators use different point masses to account for the dominant gravity fields during the spacecraft's trajectory. Two OpenGL plots are used to track the path at the starting point and ending point of the mission. Finally, four coordinate systems are used to define the spacecraft's position - based, accordingly, on Earth, Mars, the ecliptic, and the equatorial plane of the Earth.

Objects Required

- Spacecraft: MarsSupply
- Impulsive Burns: TOI, MarsMCC, MarsBPlane, and MarsOI
- Propagators: EarthProp, SunProp, and MarsProp
- Differential Corrector: MarsTOIDC
- Plots/Reports: EarthView and MarsView and a data report
- Coordinate Systems: MarsMJ2000Eq, SunMJ2kEc, SunMJ2kEq, and EarthSunRot

Creating and Modifying Objects

- Spacecraft
 - Add a spacecraft to the resource tree and rename it MarsSupply
 - Set the epoch format to TAI Mod Julian
 - Use an Epoch of 21160
 - Set the State Type to Keplerian
 - See Assumptions to modify spacecraft element properties

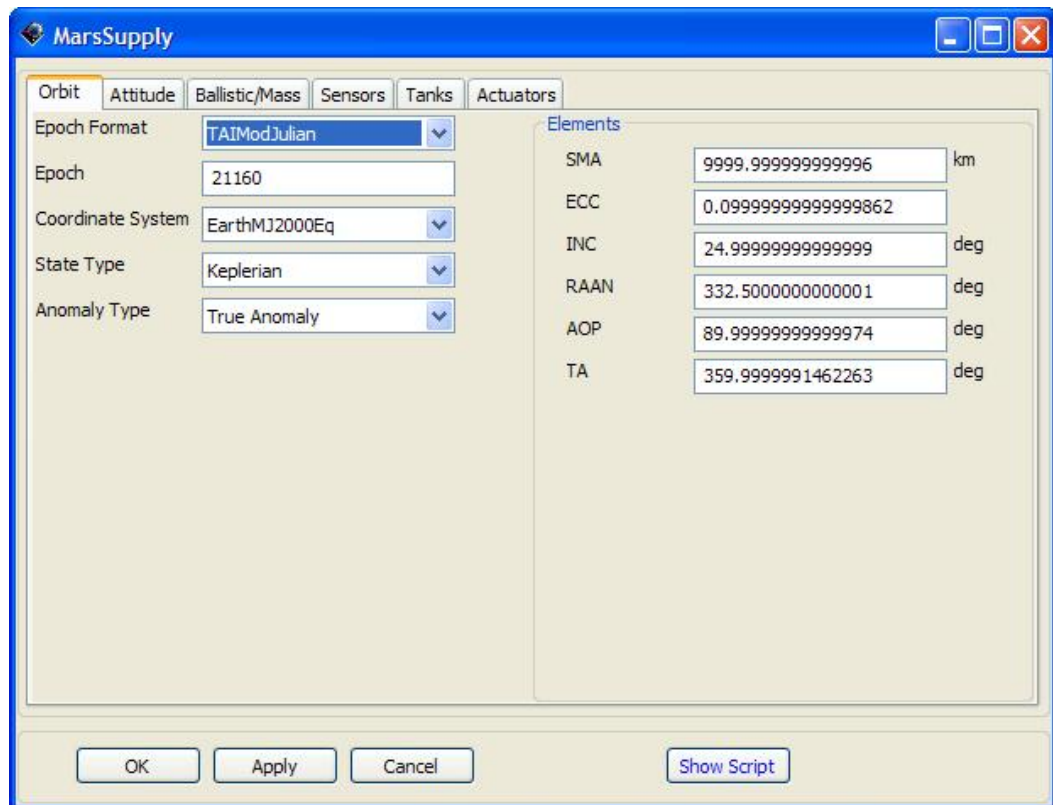


Figure: MarsSupply Spacecraft Object

- Propagators
 - Remove all primary bodies and gravity models from all propagators
 - EarthProp
 - Use an Integrator max step size of 86400 sec
 - Include only a Geocentric point mass

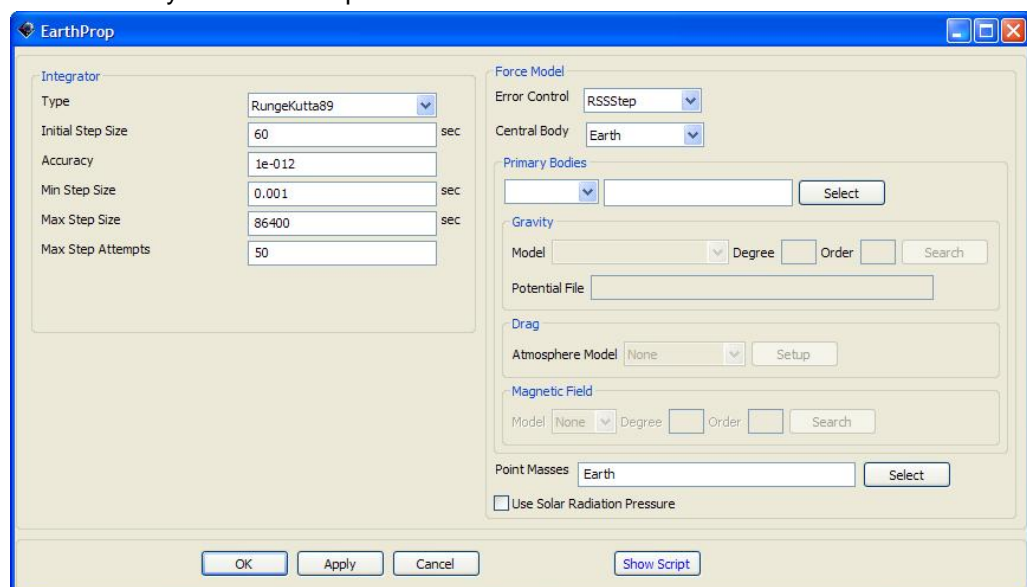


Figure: EarthProp Propagator Object

- SunProp
 - Set the Integrator max step size to 160000 sec
 - Set the Integrator max step attempts to 500
 - Include heliocentric, geocentric, and lunar point masses
- MarsProp includes only a martian point mass
 - Set the Integrator max step size to 86400 sec
 - Set the Integrator max step attempts to 100
- Burns
 - Change the TOI vector V to 2.85 km/s

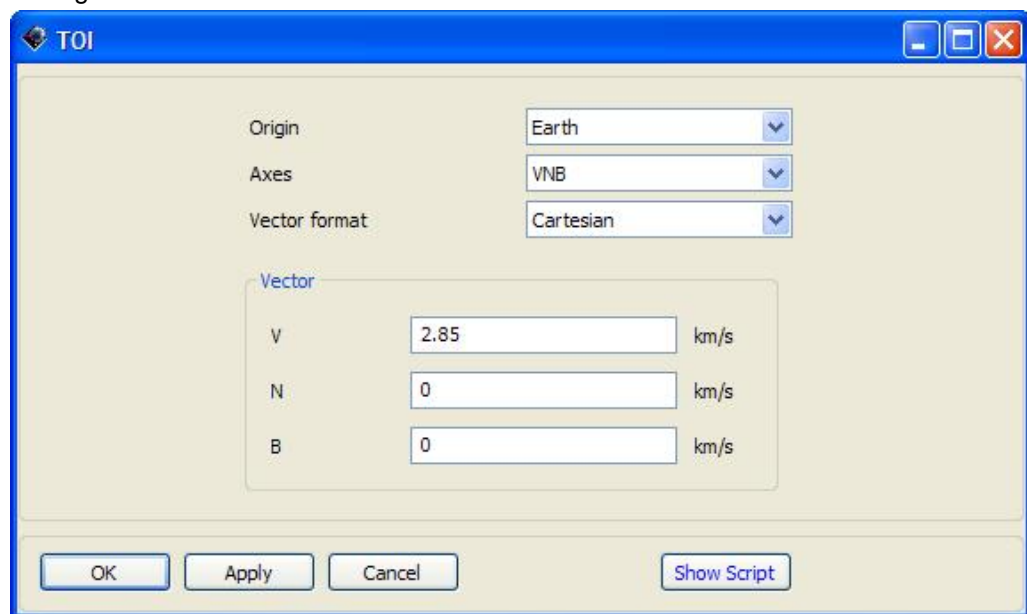


Figure: TOI Impulsive Burn Object

- Change the MarsMCC vector N to - 0.8593 km/s with a heliocentric origin
- Change the MarsBPlane vector V to - 0.3309 km/s and B to 0.6917 km/s with a heliocentric origin
- Change the MarsOI vector V to - 4 km/s with a martian origin
- Coordinate Systems
 - EarthSunRot
 - Set the Type of Axes to ObjectReferenced
 - Make Sun the Primary body and Earth the Secondary body

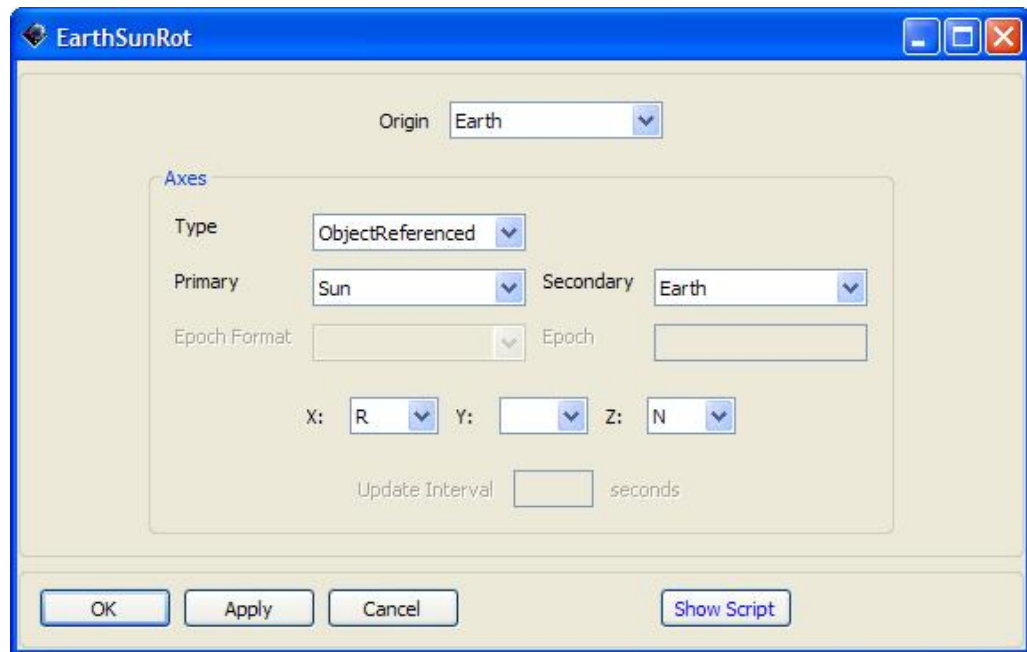


Figure: EarthSunCoord Coordinate System Object

- SunMJ2kEc
 - Apply a heliocentric origin to the system
 - Use MJ2000Ec as the Type of Axes
- SunMJ2kEq
 - Apply a heliocentric origin to the system
 - Use MJ2000Eq as the Type of Axes
- MarsMJ2000Eq
 - Apply a martian origin to the system
 - Use MJ2000Eq as the Type of Axes
- Solvers
 - Add a differential corrector to the Boundary Value Solvers folder in the Solvers folder under the resource tree and rename it MarSTOIDC
 - Use the Targeter Text File targeter_MarSTOIDC.data

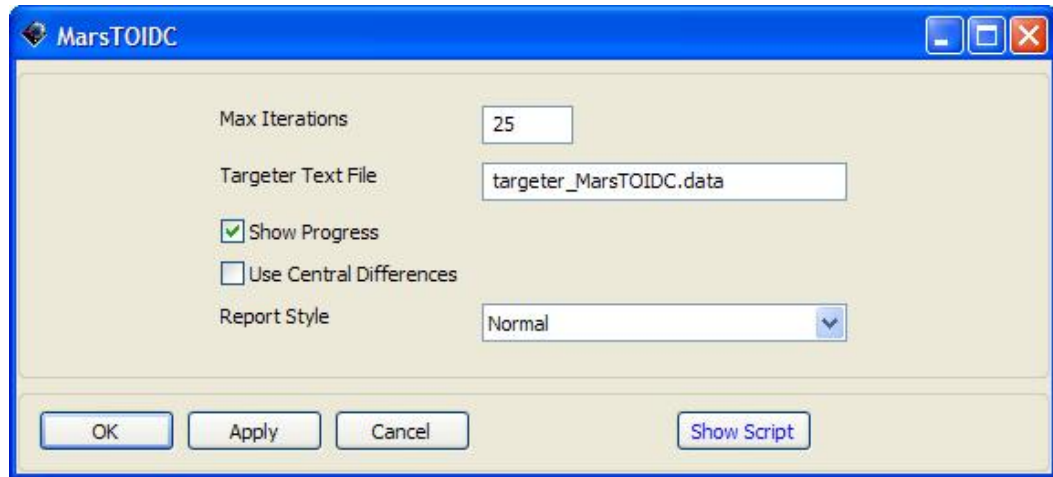


Figure: MarsTOIDC Differential Corrector Object

- Plots/Reports
- Add a Report file to the Plots/Reports folder under the resource tree

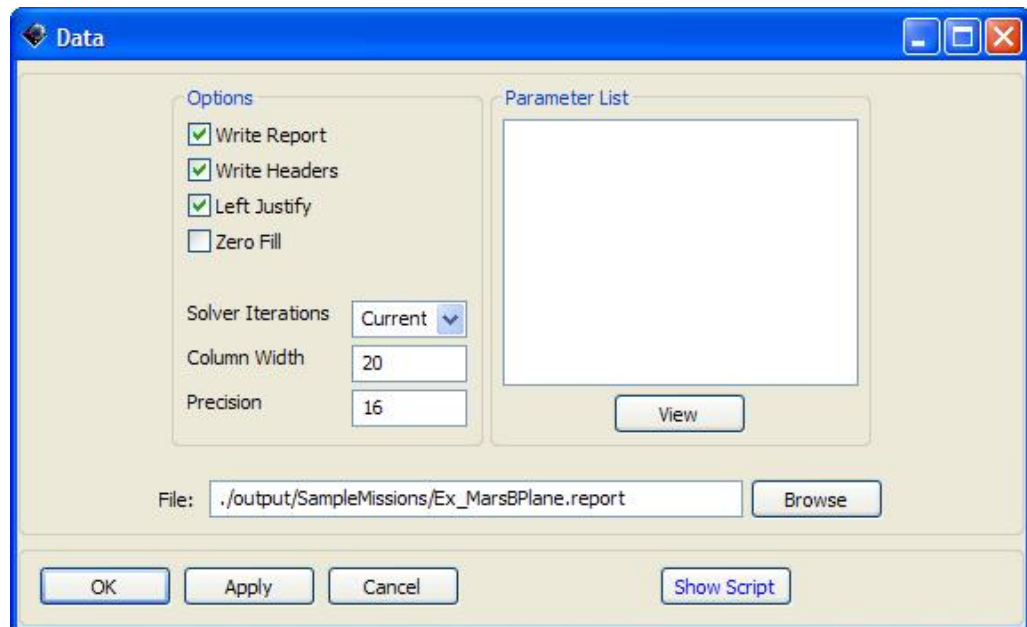


Figure: Data Data Report Object

- EarthView
 - Add the Sun to the Selected Celestial Objects field
 - Check the Draw Grid and Draw Sun Line boxes in the Drawing Option group box

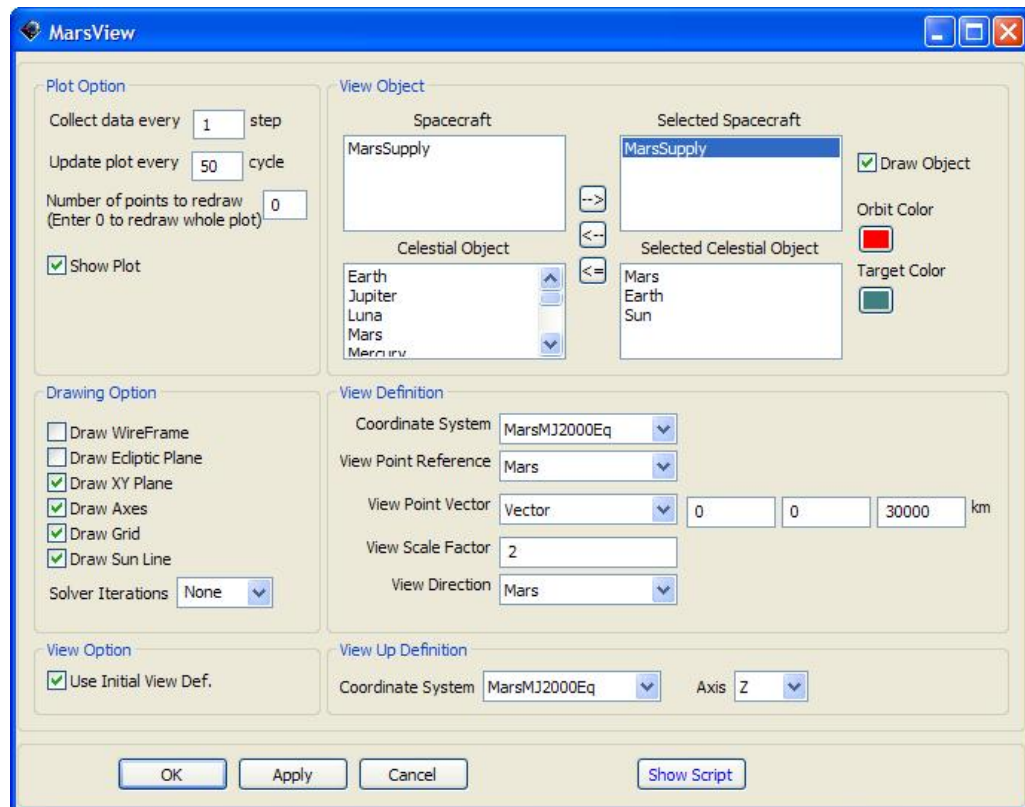


Figure: MarsView OpenGL Plot Object

- MarsView
 - Add the Sun and Mars to the Selected Celestial Objects field
 - Use MarsMJ2000Eq coordinate system in the View Definition and View Up Definition group boxes
 - Use a View Scale Factor of 2
 - Check the Draw Grid and Draw Sun Line boxes in the Drawing Option group box

Creating and Configuring the Mission Sequence

This mission begins at the spacecraft orbit's Earth periapsis point, then moves to a Mars transfer trajectory, and ends in an elliptical orbit about Mars. The transfer trajectory calls two targeting sequences, one to obtain an inclination, and the other to obtain a B-Plane position. In the first targeting sequence, the inclination is achieved, and is then used when the second sequence targets a distinct B-Plane parameter. After the second targeting sequence is accomplished, the spacecraft is slowed by a final maneuver that puts it into the elliptical orbit and continues to propagate for another half day. During each Propagate and Maneuver command before and after the targeting sequences, reports are added for tracking and analysis purposes, to monitor the spacecraft's Cartesian coordinates and velocity vectors.

- Initial Propagate and Maneuver Sequence
 - For Equation 1, adding a report automatically turns the headers on, so be sure the setting is off to avoid including unnecessary text in the report.
 - Propagate 1 simply propagates the spacecraft to Earth's periapsis.

- Maneuver 1 applies a #V in the V direction, to propel the spacecraft for 160 days through Propagate commands 2 and 3. When the propagators switch, the dominant point mass gravity field changes from Earth to Sun.
- Targeting Sequences
 - Target 1 includes Vary, Maneuver, and Achieve commands, to prompt GMAT into achieving the desired inclination. The maneuver applies the MarsMCC impulsive burn to achieve a #V in the N direction, so the spacecraft can reach 24.677° and begin its final maneuver.
 - Target 2 includes two Vary, two Propagate, two Achieve, and one Maneuver commands, to target a BdotT of 10,000 km and BdotR of – 10,000 km about Mars' periapsis. The maneuver applies the MarsBPlane impulsive burn, to achieve a #V in both the V and B velocity directions.
- Final Propagate and Maneuver Sequence
 - Maneuver 4 applies a #V in the V direction, to slow the spacecraft to the velocity that is necessary to obtain an elliptical orbit about Mars.
 - The final command, Propagate 6, propagates the spacecraft for an additional half day, so the plot can show the entire elliptical trajectory around Mars.

Creating and Modifying Sequences

Initial Propagate and Maneuver Sequence

- Report 1:
 - Add the Cartesian coordinates X, Y, and Z, and all velocity vectors, VX, VY, and VZ, to the parameters list, to report them for the MarsSupply spacecraft
 - Do not remove the A1ModJulian parameter
- Equation 1: set Data.WriteHeaders = Off
- Propagate 1:
 - Use the EarthProp propagator
 - Set the stopping condition parameter to be Earth's periapsis
- Report 2: refer to Report 1 for parameters list
- Maneuver 1: utilize the TOI burn
- Propagate 2:
 - Use the EarthProp propagator
 - Set the stopping condition to 25 elapsed days
- Report 4: refer to Report 1 for parameters list
- Propagate 3:
 - Use the SunProp propagator
 - Set the stopping condition to 135 elapsed days

- Report 5: refer to Report 1 for parameters list

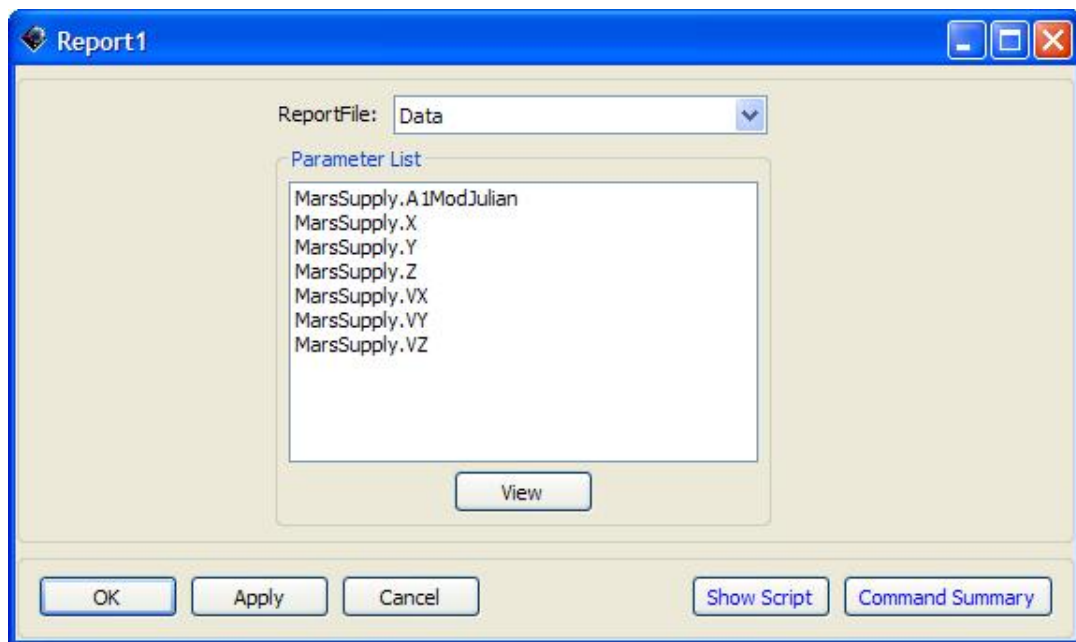


Figure: Report1 Report Command

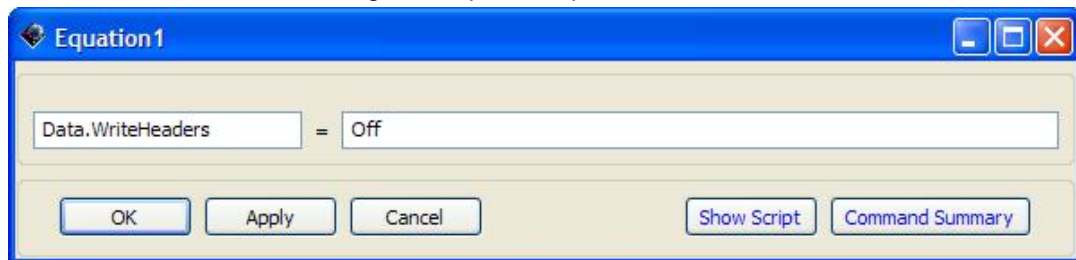


Figure: Equation1 Equation Command

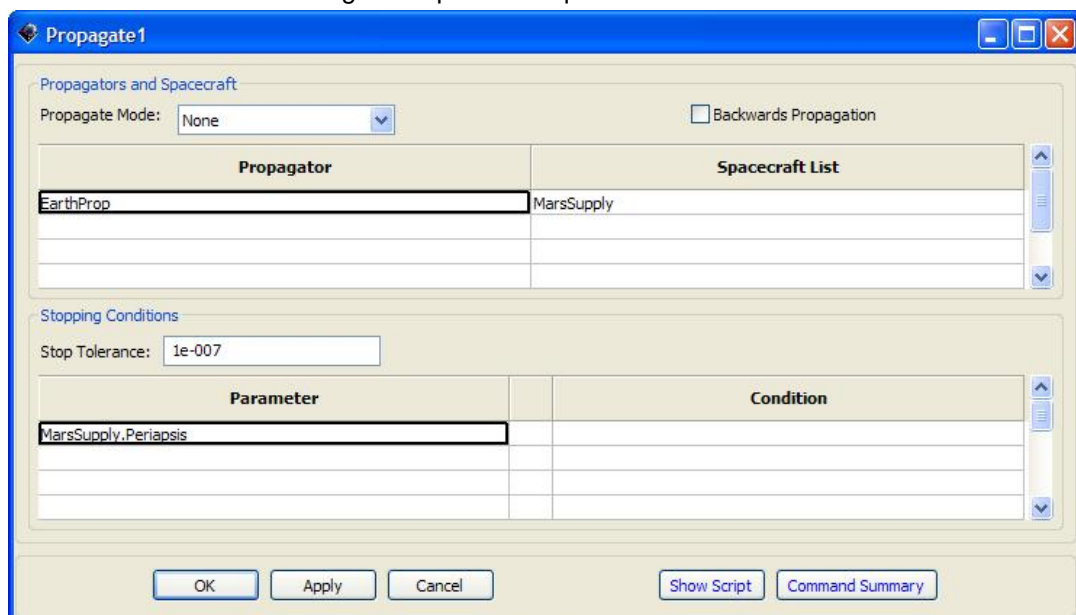


Figure: Propagate1 Propagate Command

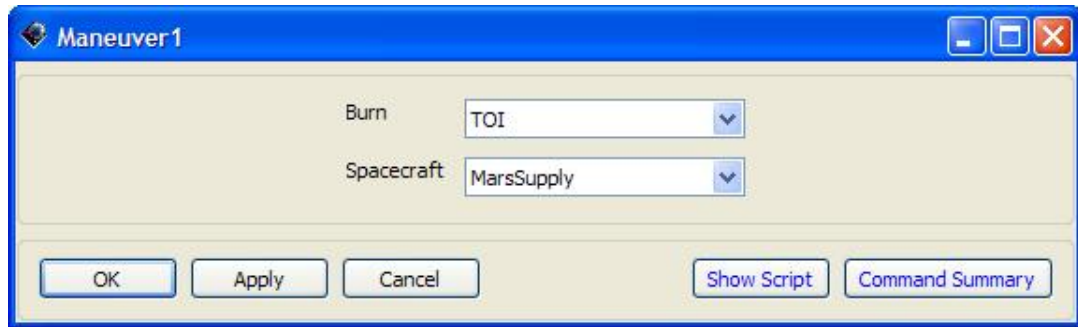


Figure: Maneuver1 Maneuver Command

Targeting Sequences

- Target 1:
 - Vary 1:
 - Set the variable to N using the MarsMCC impulsive burn
 - Set the Initial Value to 0 km/s
 - Set the Perturbation to 0.1
 - Define the Lower (Limit) to be - 8 km/s
 - Define the Upper (Limit) to be 8 km/s
 - Use a Max Step of 0.5 km/s
 - Maneuver 2: utilize the MarsMCC impulsive burn
- Achieve 1:
 - Goal:
 - Highlight MarsSupply in the Object List menu
 - Select the SunMJ2kEq coordinate system from the drop down menu
 - Add INC to the Selected Value(s) field
 - Set the Value to 24.677°
 - Use a Tolerance of 0.001°
- Report 6: refer to Report 1 for parameters list

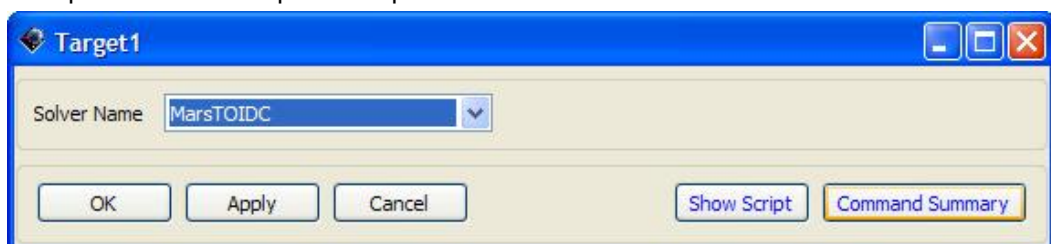


Figure: Target1 Target Command

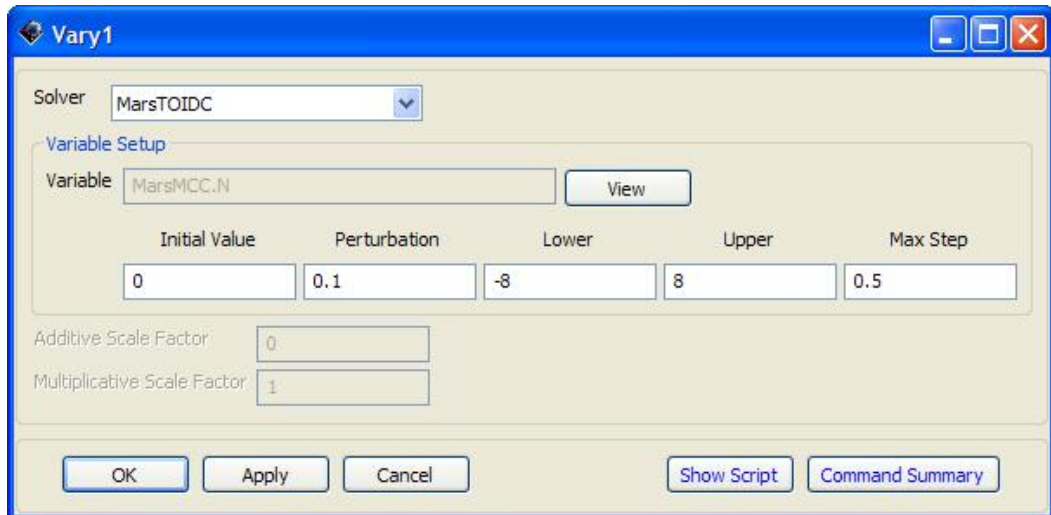


Figure: Vary1 Vary Command

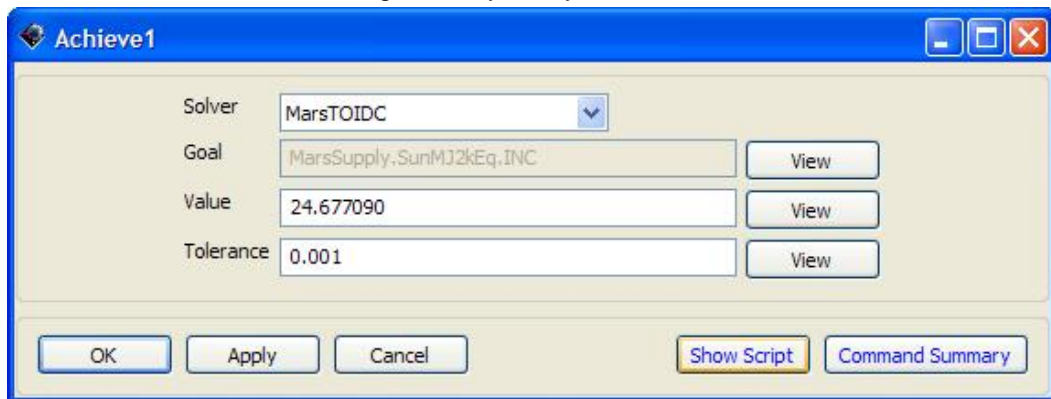


Figure: Achieve1 Achieve Command

- Target 2:
- Vary 2:
 - Set the second Variable to V using the MarBPlane impulsive burn
 - Use an Initial Value to - 0.2338 km/s
 - Change the Perturbation to 0.001
 - Define the Lower (Limit) to be - 3 km/s
 - Define the Upper (Limit) to be 3 km/s
 - Use a Max Step of 0.1 km/s
- Vary 3:
 - Set the third Variable to B using the MarsBPlane impulsive burn
 - Use an Initial Value of 0.6683 km/s
 - Change the Perturbation to 0.001
 - Define the Lower (Limit) to be - 3 km/s

- Define the Upper (Limit) to be 3 km/s
- Use a Max Step of 0.12 km/s
- Maneuver 3: Utilizes the MarsBPlane impulsive burn
- Propagate 4:
 - Use the SunProp propagator
 - Set the stopping condition parameter to 50 elapsed days
- Propagate 5:
 - Use the MarsProp propagator
 - Set the stopping condition parameter to be Mars' periapsis
- Achieve 2:
 - Goal:
 - Highlight BdotT
 - Select the MarsMJ2000Eq coordinate system from the drop down menu
 - Add BdotT to the Selected Value(s) field
 - Value = 10000 km
 - Tolerance = 50 km
- Achieve 3:
 - Goal:
 - Highlight BdotR
 - Select the MarsMJ2000Eq coordinate system from the drop down menu
 - Add BdotR to the Selected Value(s) field
 - Value = -10000 km
 - Tolerance = 50 km

Final Propagate and Maneuver Sequence

- Report 7: refer to Report 1 for parameters list
- Maneuver 4: utilize the MarsOI impulsive burn
- Report 8: refer to Report 1 for parameters list
- Propagate 6:
 - Use the MarsProp propagator
 - Set the stopping condition to 25 elapsed days
- Report 9: refer to Report 1 for the parameter list

The resulting view of Resource, Mission, and Output Trees

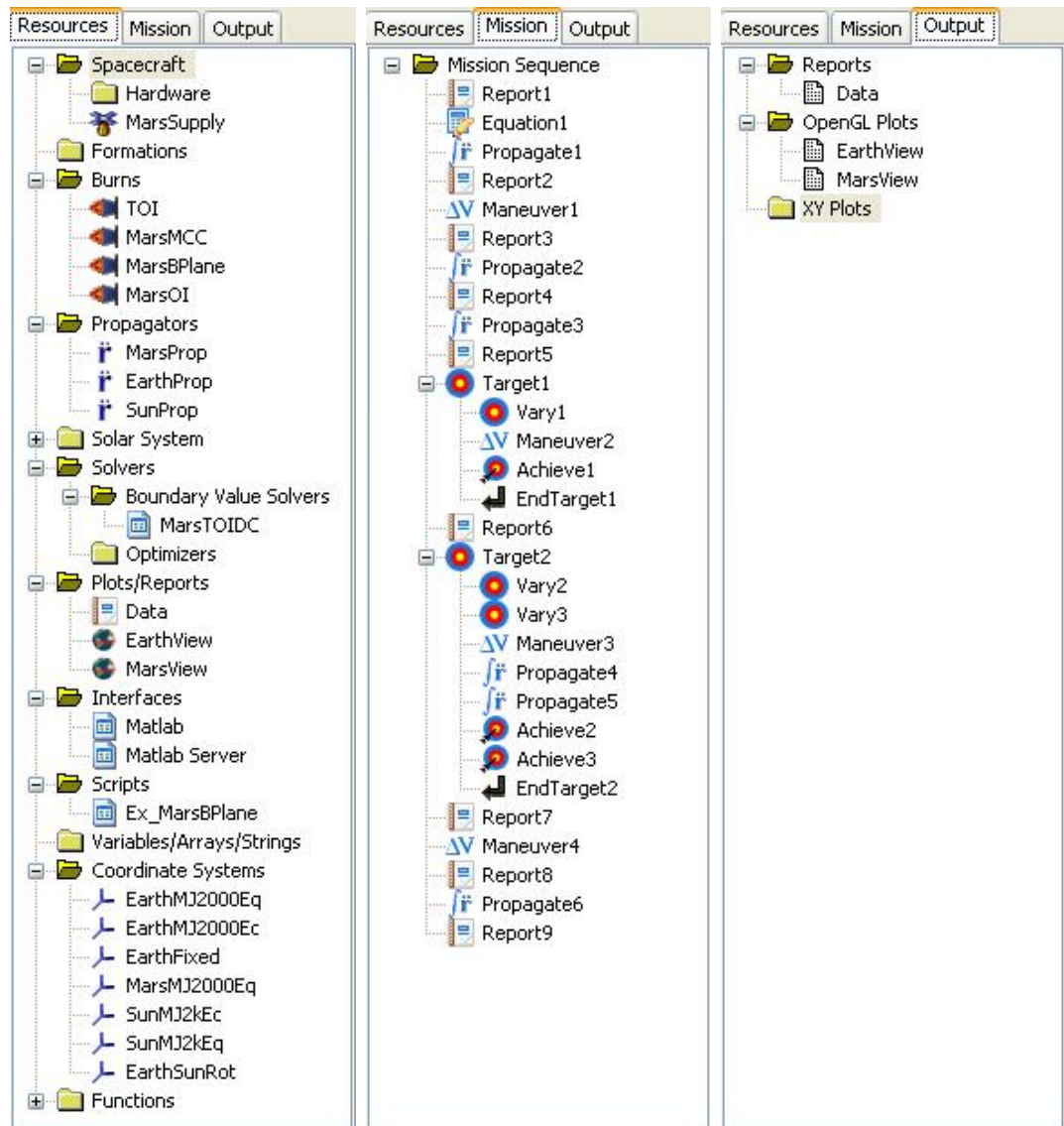


Figure: Resource, Mission, and Output Trees

Running the Mission

You have now configured the mission, and it is time to run it and see the results. First, save the changes you've made and then click the Run button (green arrow on the tool bar) to run the mission. The screen capture below summarizes what you should see after correctly completing this tutorial.

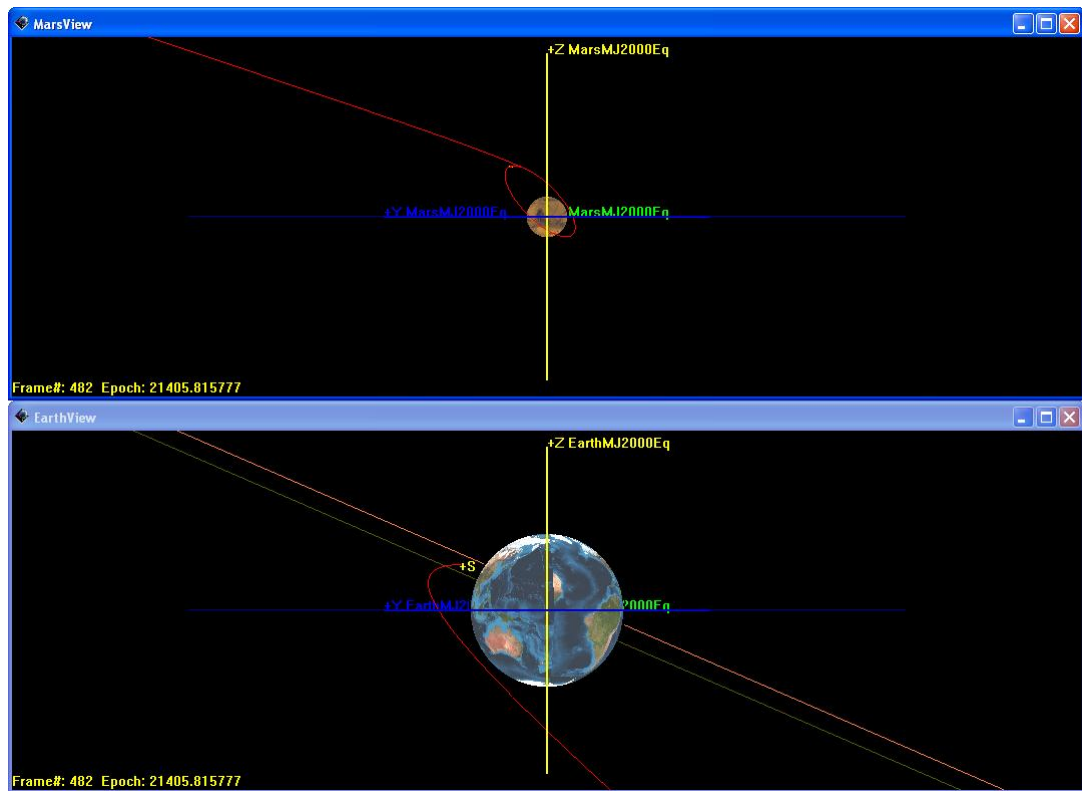


Figure: Output

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Resources

Array

Array — A two-dimensional numeric array variable

Synopsis

```
Create Array name[rows,columns];
name(row,column) = value;
...
```

Description

An array is a numeric variable that can contain multiple values in either one or two dimensions (i.e. a matrix).

Fields

\$ITEM NAME. WILL MOST LIKELY USE COURIER NEW HTML FORMATTING\$	\$ITEM DESCRIPTION\$	
	Default	\$ITEM DEFAULT VALUE\$
	Limits	\$ITEM OPTIONS\$
	Units	\$UNIT\$
\$ITEM NAME. WILL MOST LIKELY USE COURIER NEW HTML FORMATTING\$	\$ITEM DESCRIPTION\$	
	Default	\$ITEM DEFAULT VALUE\$
	Limits	\$ITEM OPTIONS\$
	Units	\$UNIT\$

Interactions

Report Commands Report commands can be used to retrieve information within arrays or from the entire array.

Examples

Example 1. Creating an array

This example creates an empty one-dimensional array with 5 elements.

```
Create Array Array1[1,5];
```

Example 2. Creating and populating a matrix

This example creates the identity matrix of size 2 and names it I:

```
Create Array I[2,2];
I(1,1) = 1;
I(1,2) = 0;
I(2,1) = 0;
I(2,2) = 1;
```

Barycenter

Barycenter — A barycenter.

Synopsis

```
Create Barycenter name
name.BodyNames = {bodyName1,bodyName2,...,bodyNameN}
```

Description

A barycenter is the center of mass of one or more celestial bodies and can be used as the origin of a `CoordinateSystem`, a reference point in an `OrbitView`, or as one of the points in a `LibrationPoint`.

Fields

Body-Names	The <code>BodyNames</code> field is a list that contains the bodies used to define a barycenter. In a script, the list must be surrounded by curly braces. (i.e. <code>BaryCenterName.BodyNames = { Earth, Luna }</code>)
Default	Earth, Luna
Limits	Sun, Mercury, Venus, Earth, Luna, Mars, Jupiter, Saturn, Uranus, Neptune, Pluto, or any user-defined celestial body. At least one body must be selected!
Units	N/A

Interactions

Coordinate Systems	Interacts with the barycenter object selecting it as the origin of the coordinate system or as a primary or secondary point for defining the axes.
OpenGL Plot	It can be selected as a celestial object to be drawn onto the plot, a View Point Reference, or the View Direction within the dialog box.

Examples

```
Create Barycenter EarthMoonBary;
GMAT EarthMoonBary.BodyNames = {Earth, Luna};
```

CelestialBodies

Celestial Bodies — A celestial body.

Synopsis

```
Create Planet name
name.field = value
```

Description

The Celestial Bodies are the main bodies of the Solar System and the Moon. They are part of the resource tree and are under Solar System. They can be selected for a large variety of options for points of reference.

Fields

Fields Associated with All Celestial Bodies

Mu	<p>The Mu field allows the user to define the gravitational parameter of a celestial body.</p> <p>Default 398600.4414</p> <p>Limits Real Number < 0</p> <p>Units km³/sec².z</p>
Equatorial Radius	<p>The EquatorialRadius field allows the user to define the equatorial radius of a celestial body.</p> <p>Default 6378.1363</p> <p>Limits Real Number < 0</p> <p>Units km</p>
Flattening	<p>The Flattening field allows the user to define the mass of a celestial body.</p> <p>Default 0.00335270</p> <p>Limits Real Number</p> <p>Units None</p>
InitialEpoch	<p>The InitialEpoch field allows the user to define the initial epoch, in A1 Modified Julian Date, for a celestial body. The initial epoch is only used when the user selects Analytic for the Ephemeris field on the solar system. In this case, GMAT solves Kepler's problem to determine the position and velocity of a celestial body, using the initial epoch and state information described below.</p> <p>Default 21544.500371</p> <p>Limits Real Number</p> <p>Units A1ModJulian</p>
SMA	<p>The SMA field allows the user to define the semimajor axis of a celestial body's orbit about its central body. (Only used when the user selects Analytic for the Ephemeris field on the Solar System.)</p> <p>Default 149653978.978377</p> <p>Limits Real Number ≥ 0</p> <p>Units km</p>
ECC	<p>The ECC field allows the user to define the eccentricity of a celestial body's orbit about its central body. (Only used when the user selects Analytic for the Ephemeris field on the Solar System.)</p>

	Default 0.017046 Limits Real Number ≥ 0 Units None
INC	<p>The INC field allows the user to define the inclination of a celestial body's orbit about its central body, in the FK5 coordinate system. (Only used when the user selects Analytic for the Ephemeris field on the Solar System.)</p> Default 23.439034 Limits Real Number Units deg
RAAN	<p>The RAAN field allows the user to define the right ascension of the ascending node of a celestialbody's orbit about its central body, in the FK5 coordinate system. (Only used when the user selects Analytic for the Ephemeris field on the Solar System.)</p> Default 0.000186 Limits Real Number Units deg
AOP	<p>The AOP field allows the user to define the argument of periapsis of a celestial body's orbit about its central body, in the FK5 coordinate system. (Only used when the user selects Analytic for the Ephemeris field on the Solar System.)</p> Default 101.741639 Limits Real Number Units deg
TA	<p>The TA field allows the user to define the true anomaly of a celestial body's orbit about its central body. (Only used when the user selects Analytic for the Ephemeris field on the Solar System.)</p> Default 358.127085 Limits Real Number Units deg

Special Fields Associated with Earth

NutationUpdate Interval	<p>The NutationUpdateInterval field, on the Earth Celestial Body, determines how often GMAT updates the Nutation matrix used in FK5 reduction. If NutationUpdateInterval is set to zero, the Nutation is updated every time a request is made to calculate the orientation of the Earth. If NutationUpdateInterval is set to a real number greater than zero, then GMAT only updates the Nutation matrix if the number of seconds defined by NutationUpdateInterval have elapsed since the last request for the Earth's orientation data.</p> Default 60 Limits Real Number ≥ 0 Units sec
--------------------------------	---

Special Fields Associated with Luna

RotationData Source	<p>The RotationDataSource, on the Luna Celestial Body, determines what source GMAT uses to obtain data describing the orientation of the moon with respect to the FK5 system. The RotationDataSource field is only used for lunar orientation data when calculating moon-based coordinate systems with the axes types of {Fixed} and {Equator}.</p>
----------------------------	---

Default	DE405
Limits	DE405, IAU2002
Units	N/A

Interactions

Coordinate System	Interacts with the celestial body in that the body can be selected as the origin of the coordinate system or as a primary or secondary point for defining the axes.
OpenGL plot	Interacts with a celestial body in a number of ways. It can be selected to be drawn onto the plot, with the color of the orbit an option. A celestial body can also be selected as the View Point Reference or the View Direction.
Propagator	Can interact with a celestial body in a number of ways. Under the Force Model heading, any celestial body can be chosen for use as the central body, as the primary bodies, and the secondary bodies.
Bary Center	Can interact with celestial bodies in that a bary center is calculated using some number of celestial bodies which can be selected.
Libration point	Can interact with celestial bodies in that a libration point is calculated using two celestial bodies which can be selected.
Spacecraft	Interacts with celestial bodies in that for a number of parameters describe how the spacecraft is positioned in relation to a celestial body. This can include the periapsis, apoapsis, and altitude, as well as a great many of other parameters.

Examples

```
Create Planet Earth;
GMAT Earth.BodyType = 'Planet';
GMAT Earth.Mass = 5.973331957140716e+024;
GMAT Earth.EquatorialRadius = 6378.1363;
GMAT Earth.Flattening = 0.0033527;
GMAT Earth.PolarRadius = 6356.75232242699;
GMAT Earth.Mu = 398600.4415;
GMAT Earth.PosVelSource = 'DE_405';
GMAT Earth.State = [ 0 0 0 0 0 0 ];
GMAT Earth.StateTime = 0;
GMAT Earth.CentralBody = 'Sun';
GMAT Earth.BodyNumber = 2;
GMAT Earth.RefBodyNumber = 3;
GMAT Earth.SourceFilename = './files/planetary_ephem/de/winp1941.405';
GMAT Earth.SourceFile = './files/planetary_ephem/de/winp1941.405';
GMAT Earth.UsePotentialFileFlag = false;
GMAT Earth.PotentialFileName = './files/gravity/earth/JGM2.cof';
GMAT Earth.AngularVelocity = [ 0 0 7.2921158553e-005 ];
GMAT Earth.HourAngle = 0;
GMAT Earth.AtmosphereModelName = '';
GMAT Earth.Order = 4;
GMAT Earth.Degree = 4;
GMAT Earth.RotationDataSource = 'NotApplicable';
GMAT Earth.NutationUpdateInterval = 60;
```

CoordinateSystem

Coordinate Systems — A coordinate system.

Synopsis

```
Create CoordinateSystem name  
name.field = value
```

Description

Coordinate Systems are critical to GMAT for several reasons. They are what every object represented within the software is referenced to. They are used by GMAT as the basis for which all calculations are made. They also provide the reference for any OpenGL Plot that is created.

Interactions

Thruster	The Thruster object allows you to set a coordinate system as its reference.
Spacecraft	In the spacecraft dialog box you may change what coordinate system the spacecraft's position is defined in reference, whatever the state type may be set as.
OpenGL Plot	Coordinate Systems are very key to the display of OpenGL Plots. They rely on coordinate systems to set how exactly the view of the plot will look using both the View Definition and View Up Definition sections of the OpenGL Plot dialog box.
Parameter Select Dialog Box	Whenever you may select a parameter using the parameter select dialog box, you have the option of selecting certain options such X, Y, Z, and several others that will require to set a coordinate system for them to reference.

Examples

```
Create CoordinateSystem EarthMJ2000Eq;  
GMAT EarthMJ2000Eq.Origin = Earth;  
GMAT EarthMJ2000Eq.Axes = MJ2000Eq;  
GMAT EarthMJ2000Eq.UpdateInterval = 60;  
GMAT EarthMJ2000Eq.OverrideOriginInterval = false;
```

DifferentialCorrector

Differential Corrector — A differential corrector.

Synopsis

```
Create DifferentialCorrector name  
name.field = value
```

Description

A differential corrector is a numerical solver for solving two-point boundary value problems. The DC in GMAT uses a simple shooting method where the derivatives are determined using finite differencing. In the mission sequence, you use the differential corrector object in a Target sequence to solve two-point value problems. For example, differential correctors are often used to determine the maneuver components required to achieve desired orbital conditions, say, B-plane conditions at a planetary flyby.

You must create and configure a differential corrector object according to your application by setting numerical properties of the solver such as tolerance and maximum iterations. You can also select among different output options that show increasing levels of information for each differential corrector iteration.

The allowable settings for a differential corrector are shown in the GUI screen shots and reference table below. You can learn more about how to use a DC in a targeting sequence by reading the help files for Target, Vary, and Achieve.

Fields

MaximumIterations	<p>The MaximumIterations field allows the user to set the maximum number of iterations the differential corrector is allowed during the attempt to find a solution. If the maximum iterations is reached, GMAT exits the target loop and continues to the next command in the mission sequence. In this case, the objects retain their states as of the last nominal pass through the targeting loop.</p> <p>Default 25 Limits Integer >= 1 Units N/A</p>
DerivativeMethod	<p>The DerivativeMethod field allows the user to choose between one-sided and central differencing for numerically determining the Jacobian matrix.</p> <p>Default ForwardDifference Limits ForwardDifference, BackwardDifference, CentralDifference Units N/A</p>
ShowProgress	<p>When the ShowProgress field is set to true, then data illustrating the progress of the differential correction process are written to the message window. The message window is updated with information on the current control variable values and the constraint variances for both on perturbation and iteration passes. When the ShowProgress field is set to false, no information on the progress of the differential correction process is displayed.</p> <p>Default true Limits true, false</p>

	Units	N/A
ReportStyle	<p>The ReportStyle field allows the user to control the amount and type of information written to the file defined in the ReportFile field. Currently, the Normal and Concise options contain the same information: the Jacobian, the inverse of the Jacobian, the current values of the control variables, and achieved and desired values of the constraints. Verbose contains values of the perturbation variables in addition to the data for Normal and Concise. Debug contains detailed script snippets at each iteration for objects who have control variables.</p> <p>Default Normal</p> <p>Limits Normal, Concise, Verbose, Debug</p> <p>Units N/A</p>	
ReportFile	<p>The ReportFile field allows the user to specify the path and file name for the differential correction report.</p> <p>Default DifferentialCorrectorDCName</p> <p>Limits Filename consistent with OS</p> <p>Units N/A</p>	

Object and Command Interactions

The Differential Corrector does not interact directly with any resource objects.

The Differential Corrector is used in the following mission sequence commands:

- Target
- Vary
- Achieve

Examples

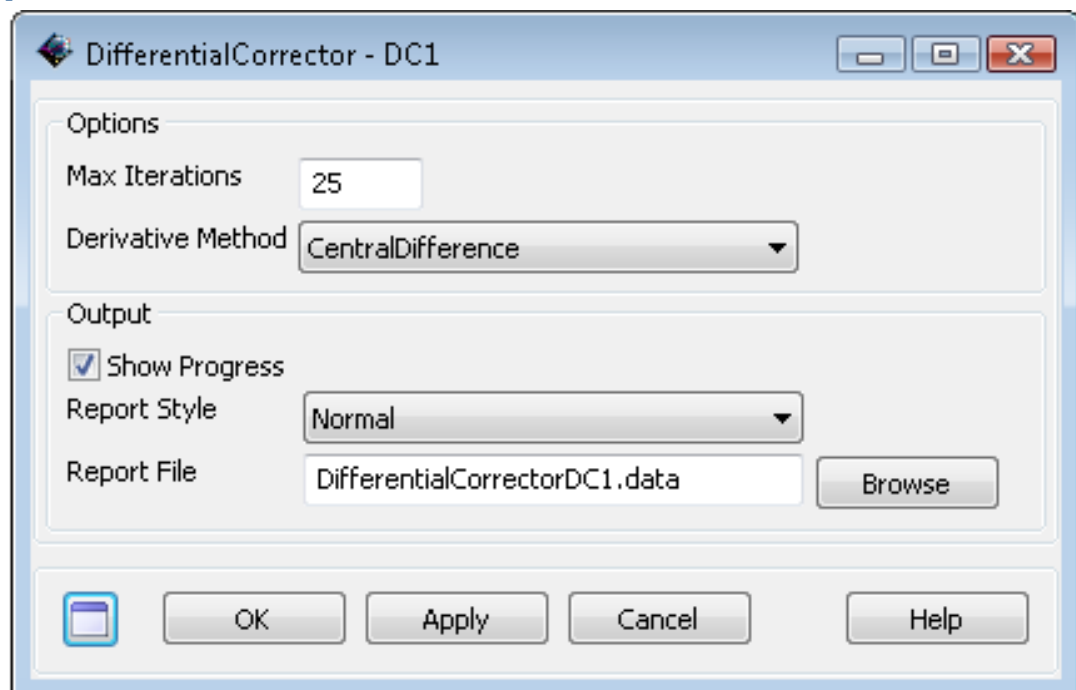


Figure: Default Name and Settings for the Differential Corrector Dialog Box

```
Create DifferentialCorrector DefaultDC;  
GMAT DefaultDC.ShowProgress = true;  
GMAT DefaultDC.ReportStyle = 'Normal';  
GMAT DefaultDC.TargeterTextFile = 'DifferentialCorrectorDefaultDC.data';  
GMAT DefaultDC.MaximumIterations = 25;  
GMAT DefaultDC.UseCentralDifferences = false;
```

EphemerisFile

EphemerisFile — An ephemeris file.

Synopsis

Under Construction.

Description

Under Construction.

Fields

Field Name	Description....
Type	Fill This In.
Default	Fill This In.
Limits	Fill This In.
Units	Fill This In.

Examples

Example 3. Example Script

%

EphemerisPropagator

EphemerisPropagator — Under Construction.

Synopsis

Under Construction.

Description

Under Construction.

Fields

Field Name	Description....
Type	Fill This In.
Default	Fill This In.
Limits	Fill This In.
Units	Fill This In.

Examples

Example 4. Example Script

%

FiniteBurn

Finite Burn — A finite burn.

Synopsis

```
Create FiniteBurn name
name.field = value
```

Description

The Finite Burn object is used when a continuous propulsion is desired. Impulsive burns happen instantaneously and through a Maneuver command, while finite burns occur until the End Finite Burn command is reached in the mission sequence and are typically coupled with Propagate commands.

Fields

Origin	<p>Together the Origin and Axes fields describe the coordinate system in which a maneuver is applied. The Origin field determines the origin of the maneuver coordinate system. The ability to define the coordinate system locally avoids having to create many coordinate systems, associated with specific spacecraft, in order to perform finite maneuvers for multiple spacecraft.</p> <p>Default Earth Limits Any celestial body, libration point, or barycenter Units N/A</p>
Axes	<p>The Axes field, together with the Origin field, describe the coordinate system in which a finite maneuver is applied. If VNB is chosen for Axes, a local coordinate system is created such that the x-axis points in the velocity direction of the spacecraft, with respect to the point defined by Origin, the y-axis points in the normal direction of the spacecraft with respect to Origin, and the z-axis completes the right-handed set.</p> <p>Default VNB Limits VNB, MJ2000Eq Units N/A</p>
Thrusters	<p>The Thrusters field allows the selection of which thrusters to use when applying a finite maneuver. The user can select more than one thruster, from the list of thrusters previously created, by including all thrusters in curly braces. An example is <code>MyFiniteBurn.Thrusters = Thruster1,Thruster2,Thruster3</code>.</p> <p>Default No Default Limits Any thruster created by user Units N/A</p>
BurnScaleFactor	<p>The BurnScaleFactor is used to scale the total acceleration before adding the acceleration due to a finite burn into the sum of the accelerations of a spacecraft. The scaling is performed by taking the sum of the accelerations applied by all thrusters specified under the Thrusters field, and multiplying the total thrust by BurnScaleFactor.</p> <p>Default 1.0 Limits Real Number Units N/A</p>

Interactions

Spacecraft	A spacecraft must be created in order to apply any burn.
Thruster	Any thruster created in the resource tree can be incorporated into a finite burn to be used on the spacecraft.
Begin and End Finite Burn command	After a finite burn is created, to apply it to the mission sequence, a Begin and End Finite Burn command must be appended to the mission tree.

Examples



Figure: Default Name and Settings for the Finite Burn Object Dialog Box

```
Create FiniteBurn FiniteBurn1;  
GMAT FiniteBurn1.Origin = Earth;  
GMAT FiniteBurn1.Axes = VNB;  
GMAT FiniteBurn1.BurnScaleFactor = 1;
```

Formation

Formation — An ephemeris file.

Synopsis

Under Construction.

Description

Under Construction.

Fields

Field Name	Description....
Type	Fill This In.
Default	Fill This In.
Limits	Fill This In.
Units	Fill This In.

Examples

Example 5. Example Script

%

FuelTank

FuelTank — A fuel tank.

Synopsis

```
Create FuelTank name
name.field = value
```

Description

A FuelTank is a thermodynamic model of a tank and is required for finite burn modelling or for impulsive burns that use mass depletion. The thermodynamic properties of the tank are modelled using the ideal gas law and assume that there is no energy transfer into or out of the tank as fuel is depleted. To use a FuelTank, you must first create the tank, and then attach it to the desired spacecraft and associate it with a thruster as shown in the examples below.

Fields

Pressure	The pressure in the tank. Type Real Number Default 1500 Limits $Pressure > 0$ Units kPa.
Temperature	The temperature of the fuel and ullage in the tank. GMAT currently assumes ullage and fuel are always at the same temperature. Type Real Number Default 20 Limits $Temperature > -273.15$ Units C.
FuelMass	The FuelMass field is the mass of fuel in the tank.

Caution

By default, GMAT will not allow the fuel mass to be negative. However, occasionally in iterative processes such as targeting, a solver will try values of a maneuver parameter that result in total fuel depletion. Using the default tank settings this will throw an exception stopping the run unless you set the AllowNegativeFuelMass flag to true.

	Type Real Number Default 756 Limits $FuelMass > 0$ Units kg.
ReferenceTemperature	The temperature of the tank when fuel was loaded. Type Real Number Default 20 Options $ReferenceTemperature \geq 0$ Units C.
Volume	The volume of the tank. GMAT checks to ensure that the volume of the tank is larger than the volume of fuel loaded in the

	tank and throws an exception in the case that the fuel volume is larger than the tank volume.
	Type Real Number
	Default 0.75
	Options Real Number > 0 such that fuel volume is < tank volume.
	Units m ³ .
FuelDensity	The density of the fuel.
	Type Real Numer
	Default 1260
	Limits Real Number > 0
	Units kg/m ³ .
PressureModel	The pressure model describes how pressure in the tank changes as fuel is depleted.
	Type Enumeration
	Default PressureRegulated
	Limits PressureRegulated, BlowDown
	Units N/A
AllowNegativeFuelMass	This field allows the fuel tank to have negagive fuel mass which can be useful in optimization and targeting sequences before convergences has occurred.
	Default false
	Options true, false.
	Units N/A

Examples

Example 6. Creating a default FuelTank and attaching it to a Spacecraft

```
% Create the Fuel Tank Object
Create FuelTank aTank;
aTank.AllowNegativeFuelMass = false;
aTank.FuelMass = 756;
aTank.Pressure = 1500;
aTank.Temperature = 20;
aTank.RefTemperature = 20;
aTank.Volume = 0.75;
aTank.FuelDensity = 1260;
aTank.PressureModel = PressureRegulated;

% Create a Thruster and assign it a FuelTank
Create Thruster aThruster;
aThruster.Tank = {aTank};

% Add the FuelTank and Thruster to a Spacecraft
Create Spacecraft aSpacecraft
aSpacecraft.Tanks = {aTank};
aSpacecraft.Thrusters = {aThruster};
```

GroundStation

GroundStation — Under Construction.

Synopsis

Under Construction.

Description

Under Construction.

Fields

Field Name	Description....
Type	Fill This In.
Default	Fill This In.
Limits	Fill This In.
Units	Fill This In.

Examples

Example 7. Example Script

%

ImpulsiveBurn

Impulsive Burn — A impulsive burn.

Synopsis

```
Create ImpulsiveBurn name
name.field = value
```

Description

The impulsive burn object in GMAT allows the spacecraft to undergo an instantaneous ΔV in up to three dimensions as opposed to a finite burn which is not instantaneous. The user can configure the burn by defining its origin, type of axes, vector format, and magnitude of the vectors. Depending on the mission, it will be simpler to use one axes or vector format over the other.

Possible Coupling with Other Objects

Spacecraft	Must be created in order to apply any burn. The purpose of the impulsive burn is to instantaneously propel the spacecraft to either target or optimize a goal during its mission.
Maneuver command	Must be created to call the burn into the mission sequence because without a maneuver, the spacecraft simply propagates around a specified trajectory. If there are several burns that exist, in the Maneuver dialog box the user can choose which burn to utilize for that part of the mission sequence. In addition, a Propagate command must follow the maneuver to allow the trajectory to unfold after a burn has been applied.
Vary command	Required a burn to be specified in the Variable Setup group box. The purpose of the Vary command is to apply a burn in order to change a parameter of the spacecraft's trajectory.

Fields

Origin	Together the Origin and Axes fields describe the coordinate system in which a maneuver is applied. The Origin field determines the origin of the maneuver coordinatesystem. The ability to define the coordinate system locally avoids having to create many coordinate systems, associated with specific spacecraft, in order to perform finite maneuvers for multiple spacecraft. Default Earth Limits Any celestial body Units N/A
Axes	The Axes field, together with the Origin field, describes the coordinate system in which an impulsive maneuver is applied. If VNB is chosen for Axes, a local coordinate system is created such that the x-axis points in the velocity direction of the spacecraft, with respect to the point defined by Origin, the y-axis points in the normal direction of the spacecraft with respect to Origin, and the z-axis completes the right-handed set. Default VNB Limits VNB, MJ2000Eq Units N/A

VectorFormat	<p>The VectorFormat field allows the user to define the format of the maneuver vector.</p> <p>Default Cartesian</p> <p>Limits Cartesian</p> <p>Units N/A</p>
Element1	<p>The Element1 field allows the user to define the first element of the impulsive maneuver vector. Element1 is X if VectorFormat is Cartesian.</p> <p>Default 0</p> <p>Limits Real Number</p> <p>Units km/sec</p>
Element2	<p>The Element2 field allows the user to define the second element of the impulsive maneuver vector. Element2 is Y if VectorFormat is Cartesian.</p> <p>Default 0</p> <p>Limits Real Number</p> <p>Units km/sec</p>
Element3	<p>The Element3 field allows the user to define the third element of the impulsive maneuver vector. Element3 is Z if VectorFormat is Cartesian.</p> <p>Default 0</p> <p>Limits Real Number</p> <p>Units km/sec</p>

Examples

```
Create ImpulsiveBurn ImpulsiveBurn1;  
GMAT ImpulsiveBurn1.Origin = Earth;  
GMAT ImpulsiveBurn1.Axes = VNB;  
GMAT ImpulsiveBurn1.VectorFormat = Cartesian;  
GMAT ImpulsiveBurn1.Element1 = 0;  
GMAT ImpulsiveBurn1.Element2 = 0;  
GMAT ImpulsiveBurn1.Element3 = 0;
```

LibrationPoint

Libration Point — A libration point.

Synopsis

```
Create LibrationPoint name
name.field = value
```

Description

A Libration point, also called a Lagrange point, is a point of equilibrium in the restricted three-body problem.

Fields

Primary	The Primary field allows you to define the body treated as the primary in the calculation of the libration point location. (See Math. Spec for more details). Default Sun Limits Sun, Mercury, Venus, Earth, Luna, Mars, Jupiter, Saturn, Uranus, Neptune, Pluto, or any Barycenter. (The Primary and Secondary bodies cannot be the same) Units N/A
Secondary	The Secondary field allows you to define the body treated as the secondary in the calculation of the libration point location. Default Earth Limits Sun, Mercury, Venus, Earth, Luna, Mars, Jupiter, Saturn, Uranus, Neptune, Pluto, or any Barycenter. (The Primary and Secondary bodies cannot be the same) Units N/A
Point	The Point field specifies which libration point the object corresponds to. Default L1 Limits L1, L2, L3, L4, L5 Units N/A

Examples

Script Syntax

```
Create Libration Point;
.Primary = ;
.Secondary = ;
.Point = <'L1', 'L2', 'L3', 'L4', 'L5'>
```

Sample Script

```
Create LibrationPoint Libration1;
GMAT Libration1.Primary = Sun;
GMAT Libration1.Secondary = Earth;
GMAT Libration1.Point = 'L1';
```

MATLABFunction

MATLABFunction — Under Construction.

Synopsis

Under Construction.

Description

Under Construction.

Fields

Field Name	Description....
Type	Fill This In.
Default	Fill This In.
Limits	Fill This In.
Units	Fill This In.

Examples

Example 8. Example Script

%

OpenGLPlot

OpenGL Plot — A OpenGL Plot.

Synopsis

```
Create OpenGLPlot name
name.field = value
```

Description

Without OpenGL Plots, you would have no way of visualizing your spacecraft as it traveled along its trajectory. OpenGL Plots also have a multitude of options that allow you to customize your view of the spacecraft. This makes these types of plots very useful and in most cases necessary to using GMAT.

Fields

Fields Associated with Plot Options

DataCollectFrequency	<p>The DataCollectFrequency field allows the user to define how data is collected for plotting. It is often inefficient to draw every ephemeris point associated with a trajectory. Often, drawing a smaller subset of the data still results in smooth trajectory plots, while executing more quickly. The DataCollectFrequency is an integer that represents how often to collect data and store for plotting. If DataCollectFrequency is set to 10, then Data is collected every 10 integration steps.</p> <p>Default 1</p> <p>Limits Integer ≥ 1</p> <p>Units Integration Steps</p>
UpdatePlotFrequency	<p>The UpdatePlotFrequency field allows the user to specify how often to update an OpenGL plot is updated with new data collected during the process of propagating spacecraft and running a mission. Data is collected for a plot according the value defined by DataCollectFrequency. An OpenGL plot is updated with the new data, according to the value set in UpdatePlotFrequency. If UpdatePlotFrequency is set to 10 and DataCollectFrequency is set to 2, then the plot is updated with new data every 20 (10×2) integration steps.</p> <p>Default 50</p> <p>Limits Integer ≥ 1</p> <p>Units Integration Steps</p>
NumPointsToRedraw	<p>When NumPointsToRedraw is set to zero, all ephemeris points are drawn. When NumPointsToRedraw is set to a positive integer, say 10 for example, only the last 10 collected data points are drawn. See DataCollectFrequency for explanation of how data is collected for an OpenGL plot.</p> <p>Default 0</p> <p>Limits Integer ≥ 0</p> <p>Units Integration Steps</p>
ShowPlot	<p>The ShowPlot field allows the user to turn off a plot for a particular run, without deleting the plot object, or removing it from</p>

the script. If you select true, then the plot will be shown. If you select false, then the plot will not be shown.

Default true
Limits true , false
Units N/A

Fields Associated with Viewed Objects

Add The Add subfield adds a spacecraft,celestial body, libration point,or barycenter to a plot. When creating a plot the Earth is added as a default body and may be removed by using the Remove command. The user can add a spacecraft, celestial body, libration point, or barycenter to a plot by using the name used to create the object. The GUI's Selected field is the equivalent of the script's Add field. In the event of no Add command or no objects in the Selected field, GMAT should run without the OpenGL plot and a warning message displayed in the message window. The following warning message is sufficient: OpenGL plot will be turned off. No object has been selected for plotting.

Default DefaultSC, Earth
Limits SpacecraftName CelestialBodyName LibrationPointName BarycenterName
Units N/A

Re-move The Remove subfield removes a spacecraft,celestial body, libration point, or barycenter from a plot. The user can remove any object that has been added to a plot by using the name used to add the object.

Default No Default
Limits Any object included in the Add list
Units N/A

Fields Associated with Drawing Options

WireFrame When the WireFrame field is set to On, celestial bodies are drawn using a wireframe model. When the WireFrame field is set to Off, then celestial bodies are drawn using a full map.

Default Off
Limits On , Off
Units N/A

EclipticPlane The EclipticPlane field allows the user to tell GMAT to draw a grid representing the ecliptic plane in an OpenGL plot. Note, the ecliptic plane can currently only be drawn for plots whose coordinate system uses the MJ2000Eq axis system.

Default Off
Limits On , Off Note: Only allowed for OpenGL plots with Coordinate Systems that use the MJ2000Eq axis system
Units N/A

XYPlane The XYPlane flag allows the user to tell GMAT to draw a grid representing the XY-plane of the coordinate system selected under the CoordinateSystem field of the OpenGL plot.

Default On
Limits On , Off
Units N/A

Axes The Axis flag allows the user to tell GMAT to draw the Cartesian axis system associated with the coordinate system selected under the CoordinateSystem field of an OpenGL plot.

Default On

	Limits	On , Off
	Units	N/A
Grid	The Grid flag allows the user to tell GMAT to draw a grid representing the longitude and latitude lines celestial bodies added to an OpenGL plot.	
	Default	On
	Limits	On , Off
	Units	N/A
EarthSunLines	The EarthSunLines allows the user to tell GMAT to draw a line that starts at the center of Earth and points towards the Sun.	
	Default	On
	Limits	On , Off
	Units	N/A
SolverIterations	The SolverIterations field determines whether or not perturbed trajectories are plotted during a solver (Targeter, Optimize) sequence. When SolverIterations is set to On, solver iterations are shown on the plot. When SolverIterations is Off, the solver iterations are not shown on the plot.	
	Default	Off
	Limits	On , Off
	Units	N/A

Fields Associated with View Definition

CoordinateSystem	The CoordinateSystem field on an OpenGL plot allows the user to select which coordinate system to use to draw the plot data. A coordinate system is defined as an origin and an axis system, and the CoordinateSystem field allows the user to determine the origin and axis system of an OpenGL plot. See the CoordinateSystem object fields for information of defining different types of coordinate systems.	
	Default	EarthMJ2000Eq
	Limits	Any default or user defined coordinate system
	Units	N/A
ViewPointReference	The ViewPointReference field is an optional field that allows the user to change the reference point from which ViewPointVector is measured. ViewPointReference defaults to the origin of the coordinate system for the plot. A ViewPointReference can be any spacecraft, celestial body, libration point, or barycenter.	
	Default	Earth
	Limits	SpacecraftName, CelestialBodyName, Libration-PointName, BarycenterName, or a 3-vector of numerical values
	Units	N/A
ViewPointVector	The product of ViewScaleFactor and ViewPointVector field determines the view point location with respect to ViewPointReference. ViewPointVector can be a vector, or any of the following objects: spacecraft, celestial body, libration point, or barycenter. The location of the Viewpoint in three-space is defined as the vector addition of ViewPointReference, and the vector defined by product of ViewScaleFactor and ViewPointVector in the coordinate system chosen by the user.	
	Default	[0 0 30000]

	Limits	SpacecraftName, CelestialBodyName, Libration-PointName, BarycenterName, or a 3-vector of numerical values
	Units	km or N/A
ViewScaleFactor	The ViewScaleFactor field scales ViewPointVector before adding it to ViewPointReference. The ViewScaleFactor allows the user to back away from an object to fit in the field of view.	
	Default	1
	Limits	Real Number ≥ 0
	Units	N/A
ViewDirection	The ViewDirection field allows the user to select the direction of view in an OpenGL plot. The user can specify the view direction by choosing an object to point at such as a spacecraft, celestial body, libration point, or barycenter. Alternatively, the user can specify a vector of the form [x y z]. If the user specification of ViewDirection, ViewPointReference, and ViewPointVector, results in a zero vector, GMAT uses [0 0 10000] for ViewDirection.	
	Default	Earth
	Limits	SpacecraftName, CelestialBodyName, Libration-PointName, BarycenterName, or a 3-vector of numerical values
	Units	km or N/A

Fields Associated with View Options

UseInitialView	The UseInitialView field allows the user to control the view of an OpenGL plot between multiple runs of a mission sequence. The first time a specific OpenGL plot is created, GMAT will automatically use the view as defined by the fields associated with View Definition, View Up Direction, and Field of View. However, if the user changes the view using the mouse, GMAT will retain this view upon rerunning the mission if UseInitialView is set to false. If UseInitialView is set to true, the view for an OpenGL plot will be returned to the view defined by the initial settings.	
	Default	On
	Limits	On , Off
	Units	N/A

Fields Associated with View Up Definition

ViewUpCoordinate System	The ViewUpCoordinateSystem and ViewUpAxis fields are used to determine which direction appears as up in an OpenGL plot and together with the fields associated the View Direction, uniquely define the view. The fields associated with the View Definition allow the user to define the point of view in 3-space, and the direction of the line of sight. However, this information alone is not enough to uniquely define the view. We also must provide how the view is oriented about the line of sight. This is accomplished by defining what direction should appear as the up direction in the plot and is configured using the ViewUpCoordinateSystem field and the ViewUpAxis field. The ViewUpCoordinateSystem allows the user to select a coordinate system to define the up direction. Most of the time this system will be the same as the coordinate system chosen under the CoordinateSystem field.	
--------------------------------	--	--

Default	EarthMJ2000Eq
Limits	Any default or user defined coordinate system
Units	N/A

ViewUpAxis

The ViewUpAxis allows the user to define which axis of the ViewUpCoordinateSystem that will appear as the up direction in an OpenGL plot. See the comments under ViewUpCoordinateSystem for more details of fields used to determine the up direction in an OpenGL plot.

Default	Z
Limits	X , -X , Y , -Y , Z , -Z
Units	N/A

Interactions

Spacecraft	Any spacecraft in your mission is available to the OpenGL Plot for display
Solar System	<p>The Sun and all of the Planets may be plotted or referenced in the OpenGL Plot</p> <p>If you add any Barrycenters or Libration Points, they will also be available for plotting and reference</p>
Coordinate Systems	Both View Definition and View Up Definition may use the three default or user added coordinate systems

Examples

```
Create OpenGLPlot DefaultOpenGL;
GMAT DefaultOpenGL.SolverIterations = Current;
GMAT DefaultOpenGL.Add = {DefaultSC, Earth};
GMAT DefaultOpenGL.OrbitColor = [ 255 32768 ];
GMAT DefaultOpenGL.TargetColor = [ 8421440 0 ];
GMAT DefaultOpenGL.CoordinateSystem = EarthMJ2000Eq;
GMAT DefaultOpenGL.ViewPointReference = Earth;
GMAT DefaultOpenGL.ViewPointVector = [ 0 0 30000 ];
GMAT DefaultOpenGL.ViewDirection = Earth;
GMAT DefaultOpenGL.ViewScaleFactor = 1;
GMAT DefaultOpenGL.ViewUpCoordinateSystem = EarthMJ2000Eq;
GMAT DefaultOpenGL.ViewUpAxis = Z;
GMAT DefaultOpenGL.CelestialPlane = Off;
GMAT DefaultOpenGL.XYPlane = On;
GMAT DefaultOpenGL.WireFrame = Off;
GMAT DefaultOpenGL.Axes = On;
GMAT DefaultOpenGL.Grid = Off;
GMAT DefaultOpenGL.SunLine = Off;
GMAT DefaultOpenGL.UseInitialView = On;
GMAT DefaultOpenGL.DataCollectFrequency = 1;
GMAT DefaultOpenGL.UpdatePlotFrequency = 50;
GMAT DefaultOpenGL.NumPointsToRedraw = 0;
GMAT DefaultOpenGL.ShowPlot = true;
```

Propagator

Propagator — A propagator.

Synopsis

```
Create Propagator name
name.field = value
```

Description

In GMAT, a Propagator is a combination of an integrator and a force model. Hence, a Propagator contains a physical model of the space environment that is used to model the motion of a spacecraft as it moves forwards or backwards in time (VOP formulation is not currently supported). You configure a Propagator by selecting among different numerical integrators and environment models to create a Propagator appropriate to the flight regime of your spacecraft during its mission. GMAT supports numerous numerical integrators as well as Force Models like point mass and non-spherical gravity, atmospheric drag (Earth), and solar radiation pressure.

To propagate spacecraft in GMAT, you first create and configure a Propagator object in the script or in the Resource Tree. Then, in the mission sequence, you create a Propagate command, the topic of another section, and select among previously existing Propagators and Spacecraft. Hence, a Propagator is different from a Propagate command: A Propagator is a resource and is found in the GUI under the resource tree, and a Propagate Event is configured under the Mission Tree and is how you instruct GMAT to propagate spacecraft.

Interfaces

The Propagator dialog box is illustrated below and contains two group boxes: the Integrator group and the Force Model group. This section discusses the items in each group on the Propagate Panel. It will present how to configure a propagator and discuss all possible user settable fields in detail.

Integrator Group

The Integrator group allows you to select and configure a numerical integrator appropriate to your application. You select the type of numerical integrator in the `--Type--` pull-down menu. After selecting the integrator type, the fields below the `--Type--` pull-down menu dynamically configure to allow you to set relevant parameters for the selected integrator type. All integrators except for Adams-Bashforth-Moulton (ABM) are configured using the same fields. The ABM integrator has the following additional fields: `--MinIntegrationerror--` and `--NomIntegrationerror--`.

Force Model Group

The Force Model group allows you to configure a force model appropriate to the flight regime of your application. The central body of propagation and error control method are also defined here. On a Propagator, GMAT classifies all celestial bodies into two mutually exclusive categories: Primary Bodies, and Point Masses. Primary bodies can have a complex force model that includes non-spherical gravity, drag, and magnetic field. Point mass bodies only have a point-mass gravitational force.

You can add a Primary Body by clicking the Select button in the Primary Bodies group box. Once you have added a Primary Body (or multiple bodies in future versions), the pull down menu allows you to configure the force model for each Primary Body. The text box, next to

the Select button contains a list of all Primary Bodies so you can see which bodies are being treated with complex force models. In future versions, GMAT will support multiple primary bodies on a propagator allowing you to use a non-spherical gravity model for the Earth and Moon simultaneously.

Configuring certain fields in the Force Model group affects the availability of other fields. For example, if you remove all bodies from the Primary Bodies list, the Gravity Field, Atmosphere Model, and Magnetic Field groups are disabled. Similarly, in the Gravity Field group, the search button and the Model File field are only active if "Other" is selected in the --+Type+-- pull-down. In the Atmosphere Model group, the Setup button is only active when --+MSISE90+-- or --+JacchiaRoberts+-- are selected in the --+Type+-- pull-down.

GMAT allows you to define Solar flux properties if you select either the --+MSISE90+-- or --+JacchiaRoberts+-- atmosphere models. By selecting one of these models in the --+Type+-- pull-down menu in the Atmosphere Model group, the Setup button is enabled. Clicking on the Setup button brings up the panel illustrated below. Here you can input Solar flux values. GMAT does not currently support flux files though future versions will.

The screenshot shows the 'DefaultProp' dialog box with the following sections and fields:

- Integrator**
 - Type: RungeKutta89 (dropdown)
 - Initial Step Size: 60 (text box) sec
 - Accuracy: 9.999999999999999e-012 (text box)
 - Min Step Size: 0.001 (text box) sec
 - Max Step Size: 2700 (text box) sec
 - Max Step Attempts: 50 (text box)
- Force Model**
 - Error Control: RSSStep (dropdown)
 - Central Body: Earth (dropdown)
 - Primary Bodies**
 - Earth (dropdown) | Earth (text box) | Select (button)
 - Gravity**
 - Model: JGM-2 (dropdown) | Degree: 4 (text box) | Order: 4 (text box) | Search (button)
 - Potential File: .\files\gravity\earth\JGM2.cof (text box)
 - Drag**
 - Atmosphere Model: None (dropdown) | Setup (button)
 - Magnetic Field**
 - Model: None (dropdown) | Degree: (text box) | Order: (text box) | Search (button)
 - Point Masses: (text box) | Select (button)
 - ☐ Use Solar Radiation Pressure
- Buttons**
 - OK, Apply, Cancel, Show Script

Figure: Default Name and Settings for the Propagator Object Dialog Box

Fields

Force Model Group Box Fields

ErrorControl

The ErrorControl field allows you to choose how a Propagator measures the error in an integration step. The algorithm selected in the ErrorControl field is used to determine the error in the current step, and this error is compared to the value set in the Accuracy field to determine if the step has an acceptable error or needs to be improved.

All error measurements are relative error, however, the reference for the relative error changes depending upon the selection of ErrorControl. RSSStep is the Root Sum Square (RSS) relative error measured with respect to the current step. RSSState is the (RSS) relative error measured with respect to the current state. LargestStep is the state vector component with the largest relative error measured with respect to the current step. LargestState

is the state vector component with the largest relative error measured with respect to the current state. For a more detailed discussion see the GMAT Mathematical Specification. Units: N/A.

Default RSSStep
Limits RSSStep, RSSState, LargestState, LargestStep

CentralBody

The CentralBody field allows the user to select the origin for the propagation. All propagation occurs in the FK5 axes system, about the CentralBody chosen by the user. The CentralBody must be a gravitational body and so cannot be a LibrationPoint or other special point.

Default Earth
Limits Sun, Mercury, Venus, Earth, Luna, Mars, Jupiter, Saturn, Uranus, Neptune, Pluto
Units N/A

PrimaryBodies

The PrimaryBodies field is a list of all celestial bodies that are to be modelled with a force model more complex than point mass gravity. Lists are surrounded by curly braces. For each PrimaryBody, the user can choose a drag, magnetic field, and aspherical gravity model. There is a coupling between the PrimaryBodies field and the PointMasses field. A primary body can be any planet or moon not included in the PointMasses field.

Default Earth
Limits Sun, Mercury, Venus, Earth, Luna, Mars, Jupiter, Saturn, Uranus, Neptune, Pluto
Units N/A

Gravity.PrimaryBody.PotentialFile This field allows the user to define the source for the non-spherical gravity coefficients for a primary body. If a gravity file is located in the Primary Body's potential path as defined in the startup file, you only need to specify the model name and not the entire path. For example, if the JGM2 coefficients file is contained in the directory defined in the startup file by the line EARTH_POT_PATH, then you only need to specify the model name JGM2. If the model is not contained in the body's potential path, you must supply the entire path as well as the file name. If GMAT does not successfully find the file requested, it uses the default gravity model as defined in the startup file. From the GUI, only models for Earth appear if Earth is the active primary body. This is to avoid allowing the user to select a lunar potential model for the Earth. If the Other option is selected the user has the ability of selecting a gravity model file on their local computer.

Default JGM2
Limits CentralBody-based models, Other. See Comments
Units N/A

Gravity.PrimaryBody.Degree

This field allows the user to select the the degree, or number of zonal terms, in the non-spherical gravity model. Ex. Gravity.Earth.Degree = 2 tells GMAT to use only the J2 zonal term for the Earth. The value for Degree must be less than the maximum degree specified by the Model.

Default 4

	Limits Integer ≥ 0 and $<$ the maximum specified by the model, Order \leq Degree Units N/A
Gravity.PrimaryBody.Order	<p>This field allows the user to select the the order, or number of tesseral terms, in the non-spherical gravity model. Ex. Gravity.Earth.Order = 2 tells GMAT to use 2 tesseral terms. Note: Order must be greater than or equal to Degree.</p> Default 4 Limits Integer ≥ 0 and $<$ the maximum specified by the model, Order \leq Degree Units N/A
Drag	<p>The Drag field allows a user to specify a drag model. Currently, only one drag model can be chosen for a particular propagator and only Earth models are available.</p> Default N/A Limits None, JacchiaRoberts, MSISE90, Exponential Units N/A. Note: This field will be deprecated in future versions of GMAT. Currently, the Drag field and the Drag.AtmosphereModel field must be set to the same value.
Drag.AtmosphereModel	<p>The Drag.AtmosphereModel field allows a user to specify a drag model. Currently, only one drag model can be chosen for a particular propagator and only Earth models are available.</p> Default None Limits None, JacchiaRoberts, MSISE90, Exponential Units N/A
Drag.F107	<p>The Drag.F107 field allows you to set the $F_{10.7}$ solar flux value used in computing atmospheric density. $F_{10.7}$ is the solar radiation at a wavelength of 10.7 cm.</p> Default 150 Limits Real Number ≥ 0 Units $\text{W/m}^2/\text{Hz}$
Drag.F107A	<p>The Drag.F107A field allows you to set the average $F_{10.7}$ value. $F_{10.7}$ is the average of $F_{10.7}$ over one month.</p> Default 150 Limits Real Number ≥ 0 Units $\text{W/m}^2/\text{Hz}$
Drag.MagneticIndex	<p>The Drag.MagneticIndex index field allows you to set the k_p value for use in atmospheric density calculations. k_p is a planetary 3-hour-average, geomagnetic index that measures magnetic effects of solar radiation.</p> Default 3 Limits $0 \leq \text{Real Number} \leq 9$ Units N/A
PointMasses	<p>A PointMass is a planet or moon that is modelled by a point source located at its center of gravity. A PointMass body can be any planet or moon not included in the PrimaryBodies field.</p> Default None

SRP	Limits	Sun, Mercury, Venus, Earth, Luna, Mars, Jupiter, Saturn, Uranus, Neptune, Pluto
	Units	N/A
	The SRP field allows the user to include the force due to solar radiation pressure in the total sum of forces.	
	Default	Off
	Limits	On, Off
	Units	N/A

Integrator Group Box Fields

Type	The Type field is used to set the type of numerical integrator. Default RungeKutta89 Limits RungeKutta89, RungeKutta68, RungeKutta56, PrinceDormand45, PrinceDormand78, BulirschStoer, Adams-BashforthMoulton
InitialStepSize	The InitialStepSize is the size of the first attempted step by the integrator. If the step defined by InitialStepSize does not satisfy Accuracy, the integrator adapts the step according an algorithm defined in the mathematical specifications document to find an acceptable first step that meets the user's requested. Default 60 (sec) Limits Real Number Units seconds
Accuracy	The Accuracy field is used to set the desired accuracy for an integration step. When you set a value for Accuracy, GMAT uses the method selected in ErrorControl field on the Force Model, to determine a metric of the accuracy. For each step, the integrator ensures that the accuracy, as calculated using the method defined by ErrorControl, is less than the limit defined by Accuracy. If an integrator exceeds MaxStepAttempts trying to meet the requested accuracy, and error message is thrown and propagation stops. Default 1e-11 Limits Real Number >= 0 Units N/A
MinStep	The MinStep field is used to set the minimum allowable step size. Default .001 (sec) Limits Real Number > 0, MinStep <= MaxStep Units seconds
MaxStep	The MaxStep field is used to set the maximum allowable step size. Default 2700.0 (sec) Limits Real Number > 0, MinStep <= MaxStep Units seconds
MaxStepAttempts	The MaxStepAttempts field allows the user to set the number of attempts the integrator takes to meet the tolerance defined by Accuracy. Default 50 Limits Integer > 0 Units None

Fields Associated Only with Adams-Bashforth-Moulton Integrator

MinIntegrationerror	The MinIntegrationerror field is used by the ABM integrator (and other predictor-corrector integrators when implemented) as the
----------------------------	---

desired integration error to be obtained when the step size is changed. Predictor-Corrector integrators adapt step size when the obtained integration error falls outside of the range of acceptable steps, as determined by the bounds set by the MinIntegrationerror and Accuracy fields. The integrator then applies an internal calculation to recompute the step size, attempting to hit the NomIntegrationerror, and restarts the integrator.

Default 1.0e-13

Limits Real Number > 0, MinIntegrationerror < NomIntegrationerror < Accuracy

Units None

NomIntegrationerror

The NomIntegrationerror field is used by the ABM integrator (and other predictor-corrector integrators when implemented) as the desired integration error to be obtained when the step size is changed. Predictor-Corrector integrators adapt step size when the obtained integration error falls outside of the range of acceptable steps, as determined by the bounds set by the MinIntegrationerror and Accuracy fields. The integrator then applies an internal calculation to recompute the step size, attempting to hit the NomIntegrationerror, and restarts the integrator.

Default 1.0e-11

Limits Real Number > 0, MinIntegrationerror , NomIntegrationerror , Accuracy

Units None

Interactions

A Propagator Requires Other Objects/Commands of Type: Force Model (Script Only). (Note: There are slight differences in how you configure a Propagator in the script and GUI and we refer you to the script example shown in the script section for details. Effort has been made to reduce any difference between the script and GUI.)

Examples

```
Create ForceModel DefaultProp_ForceModel;
GMAT DefaultProp_ForceModel.CentralBody = Earth;
GMAT DefaultProp_ForceModel.PrimaryBodies = {Earth};
GMAT DefaultProp_ForceModel.Drag = None;
GMAT DefaultProp_ForceModel.SRP = Off;
GMAT DefaultProp_ForceModel.ErrorControl = RSSStep;
GMAT DefaultProp_ForceModel.GravityField.Earth.Degree = 4;
GMAT DefaultProp_ForceModel.GravityField.Earth.Order = 4;
GMAT DefaultProp_ForceModel.GravityField.Earth.PotentialFile = 'JGM2.cof';

Create Propagator DefaultProp;
GMAT DefaultProp.FM = DefaultProp_ForceModel;
GMAT DefaultProp.Type = RungeKutta89;
GMAT DefaultProp.InitialStepSize = 60;
GMAT DefaultProp.Accuracy = 9.999999999999999e-012;
GMAT DefaultProp.MinStep = 0.001;
GMAT DefaultProp.MaxStep = 2700;
GMAT DefaultProp.MaxStepAttempts = 50;
```

ReportFile

ReportFile — A ReportFile.

Synopsis

```
Create ReportFile name
name.field = value
```

Description

The ReportFile is a file where values and qualities of objects can be stored so that they can be viewed at a later time.

Interfaces

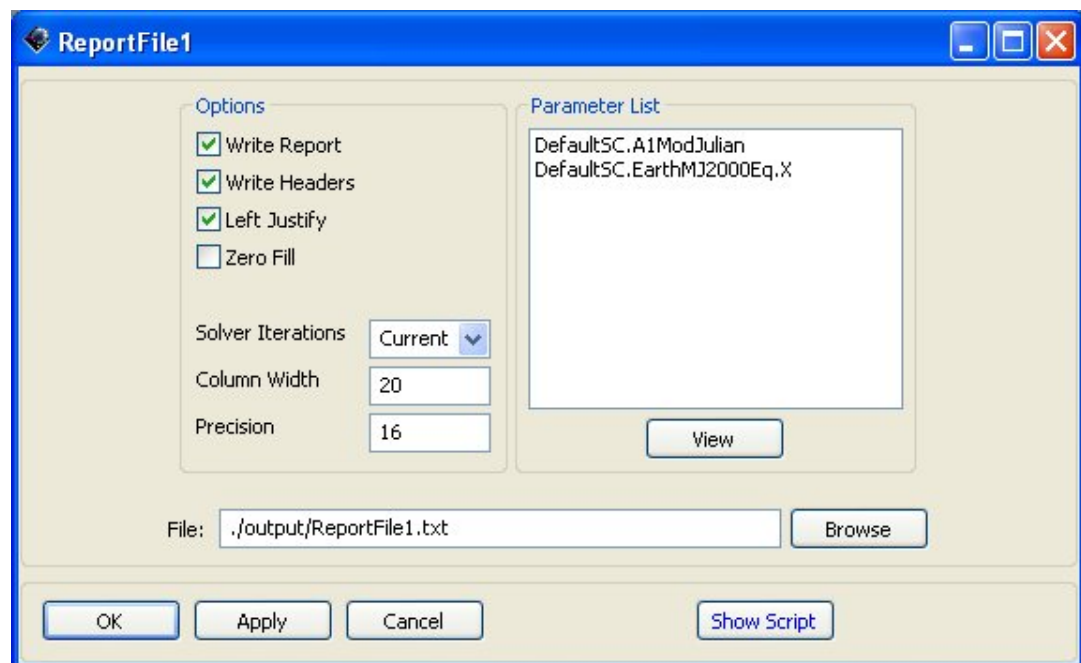


Figure: Default Name and Settings for the Report Object Dialog Box

Fields

FileName	The FileName field allows the user to define the file path and file name for a report. Default /RunReports/ReportFile1.txt Limits Valid File Path and Name Units None
Precision	The Precision field allows the user to set the precision of the variable written to a report. Default 16 Limits Integer > 0 Units Same as variable being reported.
Add	The \st{Add} field allows a user to add user-defined variables to a report file. To add multiple user-defined variables, enclose the variables with curly brackets. Ex. MyReportName.Add = {Sat.X, Sat.Y,

Var1, Array(1,1)); TheGUI's Selected field is the equivalent of the script's Add field. In the event of no Add command or no objects in the Selected field, GMAT should run without the Report output and a warning message displayed in the message window. The following warning message is sufficient: Report plot will be turned off. No object has been selected for reporting.

Default None
Limits Any user-defined parameter. Ex. Variables, Arrays, S/C parameters
Units None

WriteReport The WriteReport field specifies whether to write data to the report FileName.

Default On
Limits On, Off
Units None

WriteHeaders The WriteHeaders field specifies whether to include headers that describe the variables in a report.

Default On
Limits On, Off
Units None

LeftJustify When the LeftJustify field is set to On, then the data is left justified and appears at the left most side of the column. If the LeftJustify field is set to Off, then the data is centered in the column.

Default On
Limits On, Off
Units None

ZeroFill

Default On
Limits On, Off
Units None

ColumnWidth

The ColumnWidth field is used to define the width of the data columns in a report file. The value for ColumnWidth is applied to all columns of data. For example, if ColumnWidth is set to 20, then each data column will be 20 white-spaces wide.

Default 20
Limits Integer > 0
Units Characters

SolverIterations

The SolverIterations field determines whether or not data associated with perturbed trajectories during a solver (Targeter, Optimize) sequence is written to a report file. When SolverIterations is set to On, solver iterations are written to the report file. When SolverIterations is Off, the solver iterations are not written to the report file.

Default Off
Limits On, Off
Units None

Interactions

Report Command Located in the mission tree and will retrieve values at that particular time and insert them at the bottom of the report file.

Examples

```
Create ReportFile ReportFile1;
```

```
GMAT ReportFile1.SolverIterations = Current;  
GMAT ReportFile1.Filename = 'ReportFile1.txt';  
GMAT ReportFile1.Precision = 16;  
GMAT ReportFile1.Add = {DefaultSC.A1ModJulian, DefaultSC.EarthMJ2000Eq.X};  
GMAT ReportFile1.WriteHeaders = On;  
GMAT ReportFile1.LeftJustify = On;  
GMAT ReportFile1.ZeroFill = Off;  
GMAT ReportFile1.ColumnWidth = 20;
```

SolarSystem

Solar System — A solar system.

Synopsis

```
Create SolarSystem name
name.field = value
```

Description

This folder, found if the Solar System folder itself is double-clicked, enables the user to determine where he gets his data on the movements of planets, how often it updates, and how accurate the data is.

Fields

EphemerisSource

The EphemerisSourcefield allows the user to select the source used for planetaryephemerides. The source is used globally whenever planetaryephemeris information is required.

Default DE405

Limits DE405,DE200, SLP, Analytic

Units None

Ephemeris UpdateInterval

The EphemerisUpdateInterval is used to set howoften planetary positions are updated when calculating accelerationsduring propagation. For low-Earth orbits,EphemerisUpdateInterval can be set to around 60 for fasternumerical integration with little effect on the accuracy of thepropagation. For deep space propagation,EphemerisUpdateInterval should be set to zero.

Default 0

Limits Real Number ≥ 0

Units sec

UseTTForEphemeris

GMAT uses time in the TDB system as thedefault time system in the JPL ephemeris files. However, often it is possible to use time in the TT time system, without significantdifference in propagation accuracy. (TT and TDB are within 1millisecond of each other). The advantage to using TT is that itavoids the transformation from TT to TDB and therefore orbitpropagation will execute faster. The UseTTForEphemeris fieldallows the user to choose between the default of TDB in theephemeris files (UseTTForEphemeris=false), or TT in theephemeris files (UseTTForEphemeris = true).

Default false

Limits true, false

Units None

EphemerisFile

The EphemerisFile field allows the user to specify the locationand name of the file for each type of ephemeris GMAT supports. Forexample, if Ephemeris is set to

DE405, you can set the path for a DE405 file using `SolarSystem.EphemerisFile = c:/MyPath/MyDE405.file`.

Default Same as startup file.

Limits Filepath and file name consistent with operating system

Units None

AnalyticModel

Default LowFidelity

Limits LowFidelity

Units None

Interactions

CelestialBodies, BaryCenter, and Libration Point

The position and data on all these depend on the source of the Solar System data and how often it is updated.

Propagator

How often the position of the planetary bodies are updated will have an impact on how a spacecraft will propagate.

Spacecraft

A number of parameters of a spacecraft are based off the position of the planets

Spacecraft

Spacecraft — A spacecraft

Synopsis

```
Create Spacecraft name
name.field = value
...
```

Description

A Spacecraft resource contains information about the spacecraft's orbit, its attitude, its physical parameters (such as mass and drag coefficient), and any attached hardware, including thrusters and fuel tanks. It also contains information about the visual model used to represent it in an OrbitView.

Fields

Epoch

DateFormat The entry format and time system of the Epoch field.

Caution

The definition of the modified Julian date is not the same as other software. Most software uses the Smithsonian Astrophysical Observatory definition of 1957, where JD is the full Julian date:

$$MJD = JD - 2400000.5$$

GMAT, however, uses the following definition:

$$MJD = JD - 2430000.0$$

Value	Description
AlGregorian	A.1 time scale, Gregorian format ("DD MMM YYYY hh:mm:ss.ddd")
AlModJulian	A.1 time scale, modified Julian format
TAIGregorian	TAI time scale
TAIModJulian	<i>Default</i>
TTGregorian	TT time scale
TTModJulian	
UTCGregorian	UTC time scale
UTCModJulian	

Epoch The initial epoch of the spacecraft's state and properties.

Type string

Default 21545 (TAIModJulian format)

Limits	04 Oct 1957 12:00:00.000 or later in all time systems (6116.0 modified Julian)
Units	days (modified Julian format only)

Orbit

CoordinateSystem The Coordinate System field allows the user to choose which coordinate system with which to define the orbit state vector. The Coordinate System field has a dependency upon the State Type field. If the coordinate system chosen by the user does not have a gravitational body at the origin, then the state types Keplerian, ModifiedKeplerian, and Equinoctial are not permitted. This is because these state types require a μ value.

Type enumeration

Values

- EarthMJ2000Eq (default)
- EarthMJ2000Ec
- EarthFixed
- any user-defined coordinate system

DisplayStateType The State Type field allows the user to configure the type of state vector that they wish to use. The State Type field has a dependency upon the Coordinate System field. If the coordinate system chosen by the user does not have a gravitational body at the origin, then the state types Keplerian, ModifiedKeplerian, and Equinoctial are not permitted. This is because these state types require a μ value.

Type enumeration

Values

- Cartesian (default)
- Keplerian
- ModifiedKeplerian
- SphericalAZFPA
- SphericalRADEC
- Equinoctial

Cartesian State

x X is the x-component of the Spacecraft state in the coordinate system chosen in the Spacecraft Coordinate System field.

Type real number

Default 7100

Limits

Units km

y Y is the y-component of the Spacecraft state in the coordinate system chosen in the Spacecraft Coordinate System field.

Type real number

Default 0

Limits

Units km

z Z is the z-component of the Spacecraft state in the coordinate system chosen in the Spacecraft Coordinate System field.

Type real number

Default 1300

Limits

Units km

vx VX is the x-component of the Spacecraft velocity in the coordinate system chosen in the Spacecraft Coordinate System field.

Type real number

Default 0
Limits
Units km

VY VY is the y-component of the Spacecraft velocity in the coordinate system chosen in the Spacecraft Coordinate System field.

Type real number
Default 7.35
Limits
Units km

VZ VZ is the z-component of the Spacecraft velocity in the coordinate system chosen in the Spacecraft Coordinate System field.

Type real number
Default 1
Limits
Units km

Keplerian State

SMA The SMA field is the spacecraft orbit's osculating Keplerian semimajor axis in the coordinate system chosen in the Spacecraft Coordinate System field. SMA must be strictly greater than 1 m or less than -1 m to avoid numerical issues in the conversions to other state types. For circular and elliptical orbits ($0 \leq ECC < 0.9999999$) SMA should only be greater than 1 m and for hyperbolic orbits ($ECC > 1.0000001$) SMA should be less than -1 m. GMAT does not support the creation of parabolic orbits.

Type real number
Default
Limits $|SMA| > 1e-3 \text{ km}$
Units km

ECC The ECC field is the spacecraft orbit's osculating eccentricity. For circular or elliptic orbits, ECC must be greater than or equal to 0.0, and less than or equal to 0.9999999 to avoid numerical issues in the conversion to other state types as the Keplerian elements are undefined for parabolic orbits. For hyperbolic orbits ECC must be greater than or equal to 1.0000001. See also the SMA description.

Type real number
Default
Limits $0.0 \leq ECC \leq 0.9999999 \text{ or } ECC \geq 1.0000001$
Units none

INC The INC field is the spacecraft orbit's osculating inclination, in degrees, with respect to the selected coordinate system.

Type real number
Default
Limits
Units degrees

RAAN The RAAN field is the spacecraft orbit's osculating right ascension of the ascending node, in degrees, with respect to the selected coordinate system.

Type real number
Default
Limits
Units degrees

AOP The AOP field is the spacecraft orbit's osculating argument of periapsis, in degrees, with respect to the selected coordinate system.

Type real number
Default

	Limits	
	Units	degrees
TA	The TA field is the spacecraft orbit's osculating true anomaly.	
	Type	real number
	Default	
	Limits	
	Units	degrees

ModifiedKeplerian State

RadPer	The RadPer field is the spacecraft orbit's osculating radius of periapsis. RadPer must be greater than zero.	
	Type	real number
	Default	
	Limits	$RadPer > 0$
	Units	km
RadApo	The RadApo field is the spacecraft orbit's osculating radius of apoapsis. RadApo must be strictly greater than or less than zero. When RadApo is negative, the orbit is hyperbolic.	
	Type	real number
	Default	
	Limits	
	Units	km
INC	See the the section called "Keplerian State" section for a description of this field.	
RAAN	See the the section called "Keplerian State" section for a description of this field.	
AOP	See the the section called "Keplerian State" section for a description of this field.	
TA	See the the section called "Keplerian State" section for a description of this field.	

SphericalAZFPA State

RMAG	The RMAG field allows the user to set the magnitude of the spacecraft's position vector.	
	Type	real number
	Default	
	Limits	
	Units	km
RA	The RA field allows the user to set the spacecraft's right ascension.	
	Type	real number
	Default	
	Limits	

	Units	degrees
DEC	The DEC field allows the user to set the spacecraft's declination.	
	Type	real number
	Default	
	Limits	
	Units	degrees
VMAG	The VMAG field allows the user to set the magnitude of the spacecraft's velocity.	
	Type	real number
	Default	
	Limits	$VMAG \geq 0$
	Units	
AZI	The AZI field allows the user to set the spacecraft's azimuth angle.	
	Type	real number
	Default	
	Limits	
	Units	degrees
FPA	The FPA allows the user to set a spacecraft's flight path angle.	
	Type	real number
	Default	
	Limits	
	Units	degrees

Examples

Example 9. Creating a default Spacecraft

```
Create Spacecraft sc;
```

SQP

SQP(fmincon) — A SQP(fmincon).

Synopsis

```
Create FminconOptimizer name  
name.field = value
```

Description

fmincon is an Nonlinear Programming solver provided in MATLAB's Optimization Toolbox. fmincon performs nonlinear constrained optimization and supports linear and nonlinear constraints. This optimizer is only available to users who have both MATLAB and MATLAB's Optimization toolbox.

GMAT contains an interface to the fmincon optimizer and it appear as if fmincon is a built in optimizer in GMAT. Field names for this object have been copied from those used in MATLAB's optimset function for consistency with MATLAB as opposed to other solvers in GMAT.

Interfaces

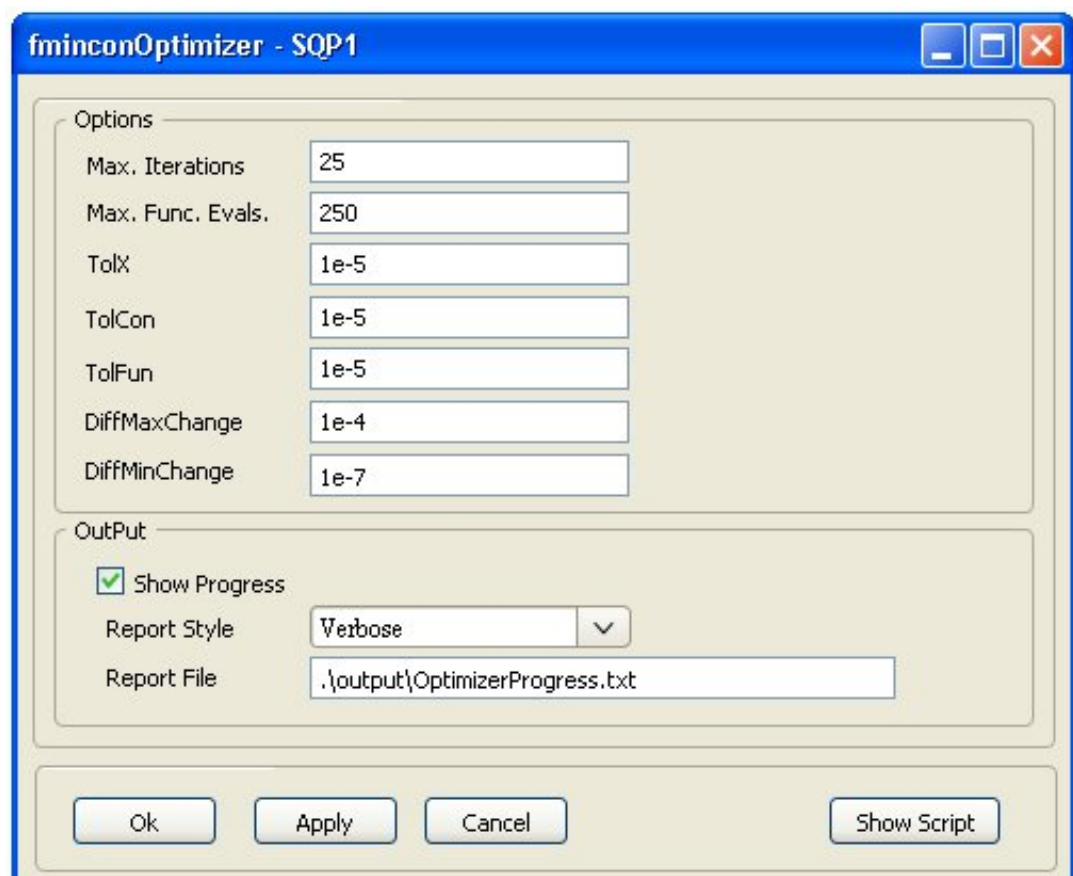


Figure: FminconOptimizer Dialog Box

Fields

DiffMax Change The DiffMaxChange parameter is the upper limit on the perturbation used in MATLAB's finite differencing algorithm. For fmincon, you don't

specify a single perturbation value, but rather give MATLAB a range, and it uses an adaptive algorithm that attempts to find the optimal perturbation.

Default 0.1
Limits Real Number > 0
Units None

DiffMin Change

The DiffMinChange parameter is the lower limit on the perturbation used in MATLAB's finite differencing algorithm. For fmincon, you don't specify a single perturbation value, but rather give MATLAB a range, and it uses an adaptive algorithm that attempts to find the optimal perturbation.

Default 1e-8
Limits Real Number > 0
Units None

MaxFunEvals

The MaxFunEvals parameter allows the user to set the maximum number of cost function evaluations in an attempt to find an optimal solution. This is equivalent to setting the maximum number of passes through an optimization loop in a GMAT script. If a solution is not found before the maximum function evaluations, fmincon outputs an ExitFlag of zero, and GMAT continues.

Default 1000
Limits Integer > 0
Units None

MaxIter

The MaxIter parameter allows the user to set the maximum allowable number of optimizer iterations. Depending upon the nature of the problem, and whether gradients are provided, it may take many function evaluations for each optimizer iteration. The MaxIter parameter allows the user to control the maximum function evaluations, and maximum iterations independently.

Default 400
Limits Integer > 0
Units None

TolX

The TolX parameter is the termination tolerance on the vector of independent variables, and is used only if the user sets a value.

Default 1e-4
Limits Real Number > 0
Units None

TolFun

The TolFun parameter is the convergence tolerance on the cost function value.

Default 1e-4
Limits Real Number > 0
Units None

TolCon

The TolCon parameter is the convergence tolerance on the constraint functions.

Default 1e-4
Limits Real Number > 0
Units None

ShowProgress

The ShowProgress field determines whether data pertaining to iterations of the solver is displayed in the message window. When ShowProgress is true, the amount of information contained in the message window is controlled by the ReportStyle field.

Default true

	Limits	true, false
	Units	None
ReportStyle	The ReportStyel field determines the amount and type of data written to the message window for each iteration of the solver (When ShowProgress is true). ADD DESCRIPTIONS OF CONCISE,VERBOSE,ADN NORMAL. I CAN'T RUN THE OPTIMIZER RIGHT NOW SO I CAN'T TELL WHAT EACH SETTING DOES.	
	Default	Normal
	Limits	Normal, Concise, Verbose, Debug
	Units	None
ReportFile	The ReportFile field contains the path and file name of the report file.	
	Default	Normal
	Limits	.\output\OptimizerData.txt
	Units	None

Using an fminconOptimizer

- Optimize Command
- Minimize
- Nonlinear Constraint

Examples

```
Create FminconOptimizer SQP1;  
GMAT SQP1.MaxIter      = 25;  
GMAT SQP1.MaxFunEvals  = 250;  
GMAT SQP1.TolX         = 1e-5;  
GMAT SQP1.TolFun       = 1e-5;  
GMAT SQP1.TolCon       = 1e-5;  
GMAT SQP1.DiffMaxChange = 1e-4;  
GMAT SQP1.DiffMinChange = 1e-7;  
GMAT SQP1.ShowProgress = true;  
GMAT SQP1.ReportStyle  = 'Verbose';  
GMAT SQP1.ReportFile   = '\output\OptimizerProgress.txt';
```

String

String — A string.

Synopsis

```
Create String name  
name.field = value
```

Description

This page will show you how to create and use String objects. Stings are useful for storing characters as a set. One possible use of them is to report back a specific message at a set point in the mission sequence.

Interactions

- | | |
|-----------------------|---|
| ReportFile | You may add a string to the Parameter List in the Report Object dialog box via the ParameterSelectDialog box. |
| Report Command | Alternatively, you may add a string to the Parameter List in the Report Command dialog box |

Using the Script

Script Syntax

```
GMAT String Name = String;
```

Script Examples

```
% The following is an example String declaration  
GMAT Example = hello world;
```

Thruster

Thruster — A thruster.

Synopsis

```
Create Thruster name
name.field = value
```

Description

The Thruster uses the fuel tank and directs the thrust of the rocket engine while in space. It is used for finite burns.

Fields

CoordinateSystem	<p>The CoordinateSystem field for a thruster determines what coordinate system the orientation parameters X_Direction, Y_Direction, and Z_Direction are referenced to. This is a temporary fix in GMAT. Eventually, the user will specify the attitude of a spacecraft, and then X_Direction, Y_Direction, and Z_Direction will be referenced to the spacecraft body frame.</p> <p>Default EarthMJ2000Eq</p> <p>Limits EarthMJ2000Eq, EarthMJ2000Ec, EarthMJ2000Eq, or any user defined system</p>
Axis	<p>The Axis field allows the user to define a local coordinate system for a thruster. Note that there is a coupling between the Axis parameter and the CoordinateSystem parameter for a thruster. Only one of the two can be specified.</p> <p>Default VNB</p> <p>Limits InertialVNB</p> <p>Units None</p>
Origin	<p>The Origin field allows the user to define a local origin for a thruster. Note that there is a coupling between the Origin parameter and the CoordinateSystem parameter for a thruster. Only one of the two can be specified.</p> <p>Default Earth</p> <p>Limits Sun, Mercury, Venus, Earth, Luna, Mars, Jupiter, Saturn, Uranus, Neptune, Pluto</p> <p>Units None</p>
X_Direction	<p>X_Direction, divided by the RSS of the three direction components, forms the x direction of the spacecraft thrust vector direction.</p> <p>Default 1</p> <p>Limits Real Number</p>
Y_Direction	<p>Y_Direction, divided by the RSS of the three direction components, forms the y direction of the spacecraft thrust vector direction.</p> <p>Default 0</p> <p>Limits Real Number</p>
Z_Direction	<p>Z_Direction, divided by the RSS of the three direction components, forms the z direction of the spacecraft thrust vector direction.</p> <p>Default 0</p>

	Limits Real Number
ThrustScaleFactor	<p>ThrustScaleFactor is a scale factor that is multiplied by the thrust vector for a given thruster, before the thrust vector is added into the total acceleration.</p> <p>Default 1</p> <p>Limits Real Number > 0</p> <p>Units None</p>
Tank	<p>The Tank field specifies which tank the thruster draws propellant from.</p> <p>The constants C_i below are used in the following equation to calculate thrust F_T as a function of pressure P and temperature T</p> <p>Default None</p> <p>Limits Tank Name</p> <p>Units $F_T(P, T) = \frac{\{C_1 + C_2 P + C_3 P^2 + C_4 P^{C_5} + C_6 P^{C_7} + C_8 P^{C_9} + C_{10} C_{11}^{C_{12} P}\} (T/T_{ref})^{1 + C_{13} + C_{14} P}}{}$</p>
C1	<p>Thrust coefficient.</p> <p>Default 500</p> <p>Limits Real Number</p> <p>Units N</p>
C2	<p>Thrust coefficient.</p> <p>Default 0</p> <p>Limits Real Number</p> <p>Units N/kPa</p>
C3	<p>Thrust coefficient.</p> <p>Default 0</p> <p>Limits Real Number</p> <p>Units N/kPa</p>
C4	<p>Thrust coefficient.</p> <p>Default 0</p> <p>Limits Real Number</p> <p>Units N/kPa^{C5}</p>
C5	<p>Thrust coefficient.</p> <p>Default 0</p> <p>Limits Real Number</p> <p>Units None</p>
C6	<p>Thrust coefficient.</p> <p>Default 0</p> <p>Limits Real Number</p> <p>Units N/kPa^{C7}</p>
C7	<p>Thrust coefficient.</p> <p>Default 0</p> <p>Limits Real Number</p> <p>Units None</p>
C8	<p>Thrust coefficient.</p> <p>Default 0</p> <p>Limits Real Number</p> <p>Units N/kPa^{C9}</p>
C9	<p>Thrust coefficient.</p> <p>Default 0</p>

	Limits	Real Number
	Units	None
C10	Thrust coefficient. Default	0
	Limits	Real Number
	Units	N
C11	Thrust coefficient. Default	1
	Limits	Real Number
	Units	None
C12	Thrust coefficient. Default	0
	Limits	Real Number
	Units	1/kPa
C13	Thrust coefficient. Default	0
	Limits	Real Number
	Units	None
C14	Thrust coefficient. Default	0
	Limits	Real Number
	Units	1/kPa
	<p>The constants K_i below are used in the following equation to calculate I_{sp} as a function of pressure P and temperature T</p> $I_{sp}(P, T) = \frac{\{K_1 + K_2 P + K_3 P^2 + K_4 P^{K_5} + K_6 P^{K_7} + K_8 P^{K_9} + K_{10} K_{11}^{K_{12} P}\} (T/T_{ref})^{1 + K_{13} + K_{14} P}}{1}$	
K1	Isp coefficient. Default	2150
	Limits	Real Number
	Units	m/sec
K2	Isp coefficient. Default	0
	Limits	Real Number
	Units	m/(sec?kPa)
K3	Isp coefficient. Default	0
	Limits	Real Number
	Units	m/(sec?kPa ²)
K4	Isp coefficient. Default	0
	Limits	Real Number
	Units	m/(sec?kPa ^{K₅})
K5	Isp coefficient. Default	0
	Limits	Real Number
	Units	None
K6	Isp coefficient. Default	0

	Limits	Real Number
	Units	m/(sec?kPa ^{k7})
K7	Isp coefficient.	
	Default	0
	Limits	Real Number
	Units	None
K8	Isp coefficient.	
	Default	0
	Limits	Real Number
	Units	m/(sec?kPa ^{k9})
K9	Isp coefficient.	
	Default	0
	Limits	Real Number
	Units	None
K10	Isp coefficient.	
	Default	0
	Limits	Real Number
	Units	m/sec
K11	Isp coefficient.	
	Default	1
	Limits	Real Number
	Units	None
K12	Isp coefficient.	
	Default	0
	Limits	Real Number
	Units	1/kPa
K13	Isp coefficient.	
	Default	0
	Limits	Real Number
	Units	None
K14	Isp coefficient.	
	Default	0
	Limits	Real Number
	Units	1/kPa

Interactions

BeginFiniteBurn/EndFiniteBurn	These commands use the tank and the thruster to start a finite burn, where the delta V is not instantaneous.
Fuel Tank	This object contains the fuel used to power the thruster and subsequently the finite burn.
FiniteBurn	This takes the parameters of the tank and the thruster and apply it to a coordinate system, with a scaling method available if wanted.
Spacecraft	This is the object that the burn is applied to.

Script Examples

```
Create Thruster Thruster1;
```

```
GMAT Thruster1.Element1 = 1;
GMAT Thruster1.Element2 = 0;
GMAT Thruster1.Element3 = 0;
GMAT Thruster1.C1 = 500;
GMAT Thruster1.C2 = 0;
GMAT Thruster1.C3 = 0;
GMAT Thruster1.C4 = 0;
GMAT Thruster1.C5 = 0;
GMAT Thruster1.C6 = 0;
GMAT Thruster1.C7 = 0;
GMAT Thruster1.C8 = 0;
GMAT Thruster1.C9 = 0;
GMAT Thruster1.C10 = 0;
GMAT Thruster1.C11 = 1;
GMAT Thruster1.C12 = 0;
GMAT Thruster1.C13 = 0;
GMAT Thruster1.C14 = 0;
GMAT Thruster1.K1 = 2150;
GMAT Thruster1.K2 = 0;
GMAT Thruster1.K3 = 0;
GMAT Thruster1.K4 = 0;
GMAT Thruster1.K5 = 0;
GMAT Thruster1.K6 = 0;
GMAT Thruster1.K7 = 0;
GMAT Thruster1.K8 = 0;
GMAT Thruster1.K9 = 0;
GMAT Thruster1.K10 = 0;
GMAT Thruster1.K11 = 1;
GMAT Thruster1.K12 = 0;
GMAT Thruster1.K13 = 0;
GMAT Thruster1.K14 = 0;
GMAT Thruster1.CoordinateSystem = 'MJ2000EarthEquator';
GMAT Thruster1.ThrustScaleFactor = 1;
```

Variable

Variable — A variable.

Synopsis

```
Create Variable name
name = value
```

Description

The Variable object allows you to create and name a variable and assign to it a real number value. A variable can be used in numerous commands which allows you to customize the mission sequence to your application. In the simplest case, a variable can be defined by a simple assignment to a numeric literal. In more complex cases, a variable can be defined using an assignment that contains a complicated mathematical expression.

Interfaces

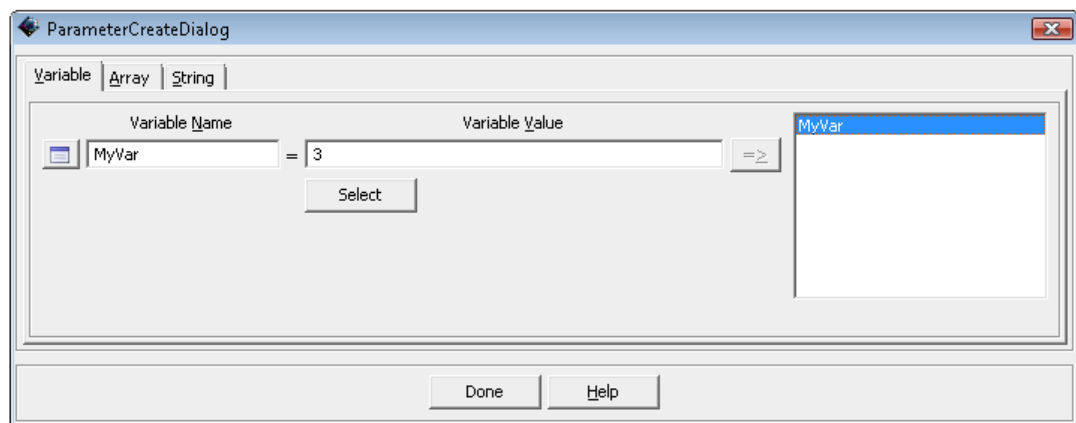


Figure: Parameter Create Dialog Box

Interactions

- Spacecraft Object
- Report Command
- Equation Command
- Vary Command
- Achieve Command
- Minimize Command

Examples

```
Create Variable pi Energy mu SMA
mu      = 398600.4415;
pi      = 3.14159265358979;
Energy  = MySpacecraft.VMAG^2/2 - mu/r;
SMA     = -mu/2/Energy;
```

VF13adOptimizer

VF13adOptimizer — Under Construction.

Synopsis

Under Construction.

Description

Under Construction.

Fields

Field Name	Description....
Type	Fill This In.
Default	Fill This In.
Limits	Fill This In.
Units	Fill This In.

Examples

Example 10. Example Script

%

XYPlot

XY Plot — A XY Plot.

Synopsis

```
Create XYPlot name
name.field = value
```

Description

The XY Plot is a graph under the Plots/Reports folder in the resource tree that plots data onto the X and Y axes of the graph. Any two parameters can be chosen to plot from the Parameter Select dialog box when the View radio button is clicked. The plot has the capability to be turned on and/or off throughout the mission if desirable.

Interactions

Spacecraft	Spacecraft interact with an XY Plot throughout the entire mission. The data retrieved from the spacecraft, as it carries out the command, is what gets plotted onto the graph.
Toggle Command	A Toggle can be inserted into the mission sequence to control when the XY Plot is to plot data by subscribing it to the Toggle list. If it is necessary to only plot data at a certain point during the mission, besides at the beginning or end points, then two Toggle commands can be added to switch it on and off.

Fields

IndVar	<p>The IndVar field allows the user to define the independent variable for an xy-plot. Only one variable can be defined as an independent variable. For example, the line <code>MyXYPlot.IndVar = DefaultSC.A1ModJulian</code> sets the independent variable to be the epoch of DefaultSC in the A1 time system and modified Julian format.</p> <table><tr><td>Default</td><td>DefaultSC.A1ModJulian</td></tr><tr><td>Limits</td><td>Any user variable, array element, or spacecraft parameter</td></tr><tr><td>Units</td><td>None</td></tr></table>	Default	DefaultSC.A1ModJulian	Limits	Any user variable, array element, or spacecraft parameter	Units	None
Default	DefaultSC.A1ModJulian						
Limits	Any user variable, array element, or spacecraft parameter						
Units	None						
Add	<p>The Add field allows the user to add dependent variables to an xy-plot. All dependent variables are plotted on the y-axis vs the independent variable defined by IndVar. To define multiple dependent variables, they should be included in curly braces. For example, <code>MyXYPlot.Add = DefaultSC.EarthMJ2000Eq.Y , DefaultSC.EarthMJ2000Eq.Z</code>. The GUI's Selected field is the equivalent of the script's Add field. In the event of no Add command or no objects in the Selected field, GMAT should run without the XYPlot and a warning message displayed in the message window. The following warning message is sufficient: XYPlot will be turned off. No object has been selected for plotting.</p> <table><tr><td>Default</td><td>DefaultSC.EarthMJ2000Eq.X</td></tr><tr><td>Limits</td><td>Any user variable, array element, or spacecraft parameter</td></tr></table>	Default	DefaultSC.EarthMJ2000Eq.X	Limits	Any user variable, array element, or spacecraft parameter		
Default	DefaultSC.EarthMJ2000Eq.X						
Limits	Any user variable, array element, or spacecraft parameter						

	Units	None
Grid	When the Grid field is set to On, then a grid is drawn on an xy-plot. When the Grid field is set to Off, then a grid is not drawn. Default On Limits On, Off Units None	
SolverIterations	The SolverIterations field determines whether or not perturbed trajectories are plotted during a solver (Targeter, Optimize) sequence. When SolverIterations is set to On, solver iterations are shown on the plot. When SolverIterations is set to Off, solver iterations are not shown on the plot. Default Off Limits On, Off Units None	
ShowPlot	The ShowPlot field allows the user to turn off a plot for a particular run, without deleting the plot object, or removing it from the script. If you select true, then the plot will be shown. If you select false, then the plot will not be shown. Default true Limits true, false Units None	

Examples

```
Create XYPlot XYPlot1;  
GMAT XYPlot1.SolverIterations = Current;  
GMAT XYPlot1.IndVar = Sat.A1ModJulian;  
GMAT XYPlot1.Add = {Sat.EarthMJ2000Eq.X};  
GMAT XYPlot1.Grid = On;  
GMAT XYPlot1.ShowPlot = true;
```

Commands

Achieve

Achieve — Perform an achieve command

Synopsis

Description

The purpose of the Achieve command is to define a goal for the spacecraft to reach at some point in its trajectory. The goal must have a corresponding value and tolerance so the differential corrector can solve for the best solution during the spacecraft's flight. To define a goal, a property must be chosen out of the Parameter Select dialog box along with the correct components in the other fields. The command can only be appended within a targeting sequence and must be accompanied and preceded by a Vary, Maneuver, and Propagate command.

Options

Goal	The option allows the user to select any single element user defined parameter, except a number, to Achieve.	
	Default	DefaultSC.Earth.RMAG
	Limits	Spacecraft parameter, Array element, Variable, or any other single element user defined parameter, excluding numbers
Arg1	The Arg1 option is the desired value for after the solver has converged.	
	Default	42165
	Limits	Real Number, Array element, Variable, or any user defined parameter that obeys the conditions of Chapter~\\ref{\\Ch:ObjectsNResources\\} for the selected
	Units	None
Tolerance	The Tolerance option sets Arg2. Arg2 is the convergence tolerance for Arg1.	
	Default	0.1
	Limits	Real Number, Array element, Variable, or any user defined parameter > 0
	Units	None
SolverName	The SolverName option allows the user to choose which solver to assign to the Achieve command.	
	Default	DefaultDC
	Limits	Any user defined differential corrector
	Units	None

Examples

```
Achieve SolverName(Goal) = Arg1, Tolerance = Arg2
```

BeginFiniteBurn

BeginFiniteBurn — Perform a begin finite burn

Synopsis

Description

The Begin Finite Burn and End Finite Burn commands are very simple. When the Begin Finite Burn command is entered into the mission sequence it will initiate the thrusters of the spacecraft until the End Finite Burn command is reached. After the finite burn is turned off, the spacecraft's thrusters will shut down.

Options

ManeuverName	The ManeuverName option allows the user to choose between any previously created finite burn. As an example, to maneuver DefaultSC using DefaultFB, the script line would appear as Maneuver DefaultFB(DefaultSC).	
	Default	DefaultFB
	Limits	Any finite burn existing in the resource tree or created in the script
SpacecraftName	Units	None
	The SpacecraftName option allows the user to select which spacecraft to maneuver using the maneuver selected in the ManeuverName option.	
	Default	DefaultSC
	Limits	Any spacecraft existing in the resource tree or created in the script
	Units	None

Examples

```
BeginFiniteBurn ManeuverName (SpacecraftName);  
EndFiniteBurn DefaultFB(DefaultSC);
```

BeginMissionSequence

BeginMissionSequence — Under construction.

Synopsis

Under construction.

Under construction.

Under construction.

Examples

Script Syntax

Under construction.

Script Examples

Under construction.

CallFunction

Call Function — Perform a call function

Synopsis

Definition

GMAT functions are very useful and work nearly the same as they do in most programming languages. They may be invoked using the Call Function command covered here.

Options

OutputList	The OutputList option allows the user to set the output of Function to a user defined parameter.	
	Default	None
	Limits	Variables, Arrays, S/C, Paramters, any other user-defined parameters, or blank. Multiple outputs must be expressed in a comma delimited list format
InputList	Units	None
	The InputList option allows the user to set the input of Function to a user defined parameter.	
	Default	None
Function	Limits	Variables, Arrays, S/C, Paramters, any other user-defined parameters, or blank. Multiple inputs must be expressed in a comma delimited list format.
	Units	None
	The Function option allows the user to set the function that will be called in a specific location of the mission sequence. The function has to be defined before it can be used in the CallFunction Command.	
	Default	None
	Limits	GMAT or Matlab Function
	Units	None

Examples

Script Syntax

```
%Function call with Inputs and Outputs
GMAT [OutputList] = Function(InputList)

%Function call with Outputs only
GMAT [OutputList] = Function
```

```
%Function call with Inputs only  
GMAT Function(InputList)  
  
Function call with no Inputs or Outputs  
GMAT Function
```

Script Examples

```
% Matlab function call without inputs or outputs  
  
% Syntax 1  
GMAT clearAll  
  
% Syntax 2  
GMAT [ ] = clearAll( )
```

Else

Else — Perform an else statement

Synopsis

Description

If-Else statements in GMAT work as they do in other programming languages, especially Matlab. The Else command adds another dimension to an If statement. You use an Else statement when you want something to happen when the conditions of an If statement are not met. For example, an If statement whose condition is "x < 5" will only execute the script within its scope when x is indeed less than 5. GMAT would otherwise pass over the If statement's associated script and continue. However, having an Else statement after the If will ensure that the lines of script within the scope of that Else are executed when x is equal to 5 or greater.

Examples

Script Syntax

```
If <logical expression>;  
    <Statements>;  
Else;  
    <Statements>;  
EndIf;
```

Script Examples

```
If DefaultSC.ElapsedDays < 1;  
    Propagate DefaultProp( DefaultSC , { DefaultSC.ElapsedDays = 0.01 } );  
Else;  
    Propagate DefaultProp( DefaultSC , { DefaultSC.ElapsedDays = 0.2 } );  
EndIf;
```

EndFiniteBurn

EndFiniteBurn — Under construction.

Synopsis

Under construction.

Under construction.

Under construction.

Examples

Script Syntax

Under construction.

Script Examples

Under construction.

Equation

Equation — Perform an equation command

Synopsis

Description

The Equation command uses the Equation to make one variable equal to some combination of previously defined variables and values. It is highly useful for storing values so that they aren't lost. Additionally, it is very useful for advanced commands.

Options

Arg1 The Arg1 option allows the user to set Arg1 to Arg2.

Default None

Limits Spacecraft Parameter, Array element, Variable, or any other single element user defined parameter

Units None

Arg2 The Arg2 option allows the user to define Arg1.

Default None

Limits Spacecraft Parameter, Array element, Variable, any other single element user defined parameter, or a combination of the aforementioned parameters using math operators

Units None

Examples

Script Syntax

```
GMAT Arg1 = Arg2;
```

Script Examples

```
% Setting a variable to a number
GMAT testVar = 24;
% Setting a variable to the value of a math statement
GMAT testVar = (testVar2 + 50)/2;
```

For

For — Perform a for loop

Synopsis

Description

The for loop is a control flow statement that allows portions of code to be executed iteratively using an explicit loop variable (Wikipedia). GMAT for loops are three-expression loops that allow the user to set the initial value of the loop variable, its increment, and the test to exit the loop. A parameter must be defined explicitly using a Create Variable statement or GUI equivalent before it can be used in a for loop command statement. The parameters used to define Start, Increment, and End can be any of the following GMAT parameters: numeric literal (real number), variable, array element, object property.

Interfaces

The GUI for the For Loop command is divided into four sections.

- The first section, the index, is where the counter variable name is displayed.
- The second section, the start, is the number with which the counter variable is first stored with.
- The third section, the increment, is the value that the counter variable will change by each time the program goes through the loop.
- The fourth section, the end, is the value of the counter variable when the loop is exited.

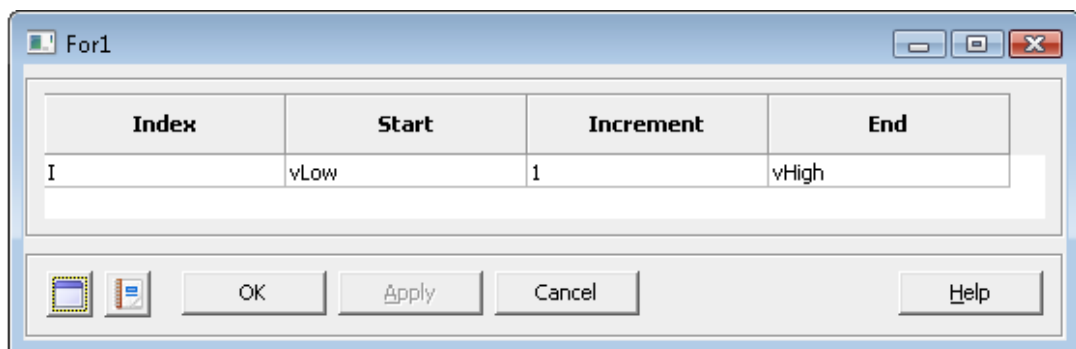


Figure: Default Name and Settings for the For Loop command Dialog Box

Script Syntax

(Simple For Loop)

```
For Variable = Start:End;  
  
EndFor; }
```

(Expanded For Loop)

```
For Variable = Start:Increment:End;
```

```
EndFor;
```

Script Examples

```
% Output the value of the For loop Variable to a file
For I = 1:1:10;
GMAT testVar = I;
Report DefaultReportFile I;
EndFor;
```

Options

Variable	The Variable field allows the user to define the variable that to be incremented during the loop process.
Default	None
Limits	Variable
Units	None
Start	The Start option allows the user to set the starting value of the For Loop.
Default	1
Limits	Real Number, Array element, Variable, or any user defined parameter
Units	None
Increment	The Increment option allows the user to set the increment value of the For Loop. When the increment value is not included in a for loop statement, the default value is used.
Default	1
Limits	Real Number, Array element, Variable, or any user defined parameter
Units	None
End	The End option allows the user to set the ending value of the For Loop.
Default	10
Limits	Real Number, Array, Variable, or any user defined parameter
Units	None

For Loop Behavior

When the Increment option is left out of the script syntax the default value is used. If an Increment value of 0 is used, the For Loop should not execute but GMAT should continue to run. If $\text{End} > \text{Start}$ and $\text{Increment} < 0$, then the For Loop should not execute. If $\text{Start} > \text{End}$ and $\text{Increment} > 0$, then the For Loop should not execute. End can be equal to Start, but the For Loop will not execute.

If

If — Perform an if command

Synopsis

Description

The If command gives you the ability to use a logical statement within GMAT. At some point during a mission sequence, when a particular command should only be executed when a certain condition is met, use of the If command is recommended. The If command also gives you the ability to make a command's execution reliant upon multiple conditions.

Options

<If Command>	Arg1 and Arg2 can be any of the following: Real Number, Array element, Variable, Spacecraft Parameter or any other user defined parameter.
	Default DefaultSC.ElapsedDays < 1.0
	Limits Arg1 < Arg2 and < can be >, <, >=, <=, ==, ~126~= Units None
<Statements>	Default None
	Limits Any script line that can be in the mission sequence
	Units None
 	The option allows the user to set an OR operator in between <logical expression>s.
	Default None
	Limits None
	Units None
&	The & option allows the user to set an AND operator in between <logical expression>s
	Default None
	Limits None
	Units None

Examples

Using the If command in the script is quite simple. If you have ever programmed before in the higher level languages such as C, Matlab, or Java, GMAT will be very familiar. The statement reads like you see it basically: If the given statement after the 'If' is true, then execute the statement(s) following until the 'EndIf' is reached.

Script Syntax

- Simple If Statement

```
If <logical expression>;  
    <Statements>;  
EndIf;
```

- Compound If statement

```
If <logical expression> | <logical expression> & <logical expression>;  
    <Statements>;  
EndIf;
```

Script Examples

```
If DefaultSC.ElapsedDays < 1;  
    Propagate DefaultProp( DefaultSC , { DefaultSC.ElapsedDays = 0.01 } );  
EndIf;  
  
If MyVariable < MyArray(1,1);  
    MyArray(1,1) = 5;  
EndIf;  
  
If DefaultSC.Earth.TA < MyArray(1,2);  
    Propagate DefaultProp( DefaultSC );  
EndIf;
```

Maneuver

Maneuver — Perform a maneuver command

Synopsis

Description

The Maneuver command is placed in the mission tree and applies a selected impulsive burn to a selected spacecraft. A finite burn requires something else to be applied.

Options

BurnName	The BurnName field allows the user to choose between any previously created impulsive burn. As an example, to maneuver DefaultSC using DefaultIB, the script line would appear as Maneuver DefaultIB(DefaultSC).
Default	DefaultIB
Limits	Any impulsive burn existing in the resource tree or created in the script
Units	None
SpacecraftName	The SpacecraftName field allows the user to select which spacecraft to maneuver using the maneuver selected in the BurnName field.
Default	DefaultSC
Limits	Any spacecraft existing in the resource tree or created in the script
Units	None

Examples

Script Syntax

```
Maneuver BurnName (SpacecraftName);
```

Script Examples

```
% Impulsive Burn  
Maneuver DefaultIB(DefaultSC);
```

Minimize

Minimize — Perform a minimize command

Synopsis

Description

The minimize command in GMAT allows variables to minimize by using a defined optimizer in optimize sequence.

Possible Coupling with Other Objects

- fmincon Object
 - Must set in order to use Minimize Command.
- Optimize Command
 - Must be defined in order to use minimize command in optimize sequence.

Options

OptimizerName	The OptimizerName option allows the user to specify which solver to use to minimize the cost function. Default SQP1 Limits Any existing fmincon solver Units None
Arg	The Arg field allows the user to specify the function to be minimized upon convergence of the solver given by OptimizerName. Arg can be any of the following: Variable, Array element, or Spacecraft Parameter or any other 1x1 numeric user defined parameter. Default DefaultSC.ECC Limits Variable, Spacecraft parameter, or Array element Units None

Examples

Script Syntax

```
Minimize OptimizerName (Arg)
```

Script Examples

```
% Minimize the eccentricity of Sat, using fminconSQP
Minimize fminconSQP(Sat.ECC);
```

```
% Minimize the Variable DeltaV, using fminconSQP
Minimize fminconSQP(DeltaV);

% Minimize the first component of MyArray, using fminconSQP
Minimize fminconSQP(MyArray(1,1));
```

NonLinearConstraint

NonLinearConstraint — Apply nonlinear constraint

Synopsis

Description

The nonlinear constraints in GMAT allows spacecraft properties, variable or array to apply constraint values, and also NonLinearConstraint can be created in optimize sequences. By using the fmincon optimizer, users can give various nonlinear constraints.

Possible Coupling with Other Objects

- Optimize Command
 - NonLinearConstraints are used in Optimize Command.
- Optimizers (Solvers)
 - Must set optimizer in order to apply NonLinearConstraints.

Options

OptimizeName	The OptimizerName option allows the user to specify which solver to use in satisfying nonlinear constraints. Default SQP1 Limits Any existing fmincon solver Units None
{logical expression}	The logical expression field allows the user to specify the constraint to be satisfied upon convergence of the solver given by OptimizerName. Arg1 and Arg2 can be any of the following: Real Number, a 1-D Array (column vector), Array element, Variable, Spacecraft Parameter or any other numeric user defined parameter. If Arg1 is a 1-D Array, then Arg2 must be a 1-D Array with the same dimensions and vice-versa. Default DefaultSC.SMA = 7000 Limits Arg1 <= Arg2 where <= can be >= ; <= ; = Units None

Examples

Script Syntax

```
NonLinearConstraint OptimizerName ({logical expression})
```

Script Examples

```
% Constrain the SMA of Sat to be 7000 km, using fminconSQP
```

```
NonLinearConstraint fminconSQP( Sat.SMA = 7000 );

% Constrain the SMA of Sat to be less than or equal to 7000 km, using fminconSQP
NonLinearConstraint fminconSQP( Sat.SMA <= 7000 );

% Constrain the SMA of Sat to be greater than or equal to 7000 km, using fminconSQP
NonLinearConstraint fminconSQP( Sat.SMA >= 7000a );
```

Optimize

Optimize — Perform an optimize command

Synopsis

Description

The optimize command in GMAT allows variables to optimize by using a solver fmincon object.

Possible Coupling with Other Objects

- fmincon Object
 - Must set in order to use Optimize Command.
- Minimize
 - Must be defined if you would like to minimize variable in an optimize sequence.
- Vary
- NonLinearConstraint

Options

SolverName	The SolverName field allows the user to choose between any previously created optimizer for use in an optimization sequence. For example, to begin a optimization sequence using DefaultSQP, the script is Optimize DefaultSQP.
Default	DefaultSQP
Limits	Any existing optimizer
Units	None
<Statements>	
Default	None
Limits	Any non-targeter and non-optimizer command lines used in the mission sequence, as well as the optimizer dependent command lines Vary, NonLinearConstraint, and Minimize.
Units	None

Examples

Script Syntax

```
Optimize SolverName;  
    <Statements>;  
EndOptimize;
```

Script Examples

```
% Beginning and ending syntax for the Optimize command  
Optimize DefaultDC;  
  
EndOptimize;
```

PenUp

PenUp — Under construction.

Synopsis

Under construction.

Under construction.

Under construction.

Examples

Script Syntax

Under construction.

Script Examples

Under construction.

PenDown

PenDown — Under construction.

Synopsis

Under construction.

Under construction.

Under construction.

Examples

Script Syntax

Under construction.

Script Examples

Under construction.

Propagate

Propagate — Perform a propagate command

Synopsis

Description

The Propagate Command is a very important one and will be covered in this section. Basically, Propagate will take the given spacecraft and, using the Propagator specified as its guide, make it travel until the given condition is met whether it be a location or something else such as elapsed time.

Reference Table

BackProp

The BackProp option allows the user to set the flag to enable or disable backwards propagation for all spacecraft in the SatListN option. The Backward Propagation GUI check box field stores all the data in BackProp. A check indicates backward propagation is enabled and no check indicates forward propagation. In the script, BackProp can be the word Backwards for backward propagation or blank for forward propagation.

Default None

Limits Backwards or None

Units None

Mode

The Mode option allows the user to set the propagation mode for the propagator that will affect all of the spacecraft added to the SatListN option. For example, if synchronized is selected, all spacecraft are propagated at the same step size. The Propagate Mode GUI field stores all the data in Mode. In the script, Mode is left blank for the None option and the text of the other options available is used for their respective modes.

Default None

Limits Synchronized or None

Units None

PropagatorName

The PropagatorName option allows the user to select a user defined propagator to use in spacecraft and/or formation propagation. The Propagator GUI field stores all the data in PropagatorName.

Default DefaultProp

Limits Default propagator or any user-defined propagator

Units None

SatListN	<p>The SatListN option allows the user to enter all the satellites and/or formations they want to propagate using the PropagatorName propagator settings. The Spacecraft List GUI field stores all the data in SatListN.</p> <p>Default DefaultSC</p> <p>Limits Any existing spacecraft or formations, not being propagated by another propagator in the same Propagate event. Multiple spacecraft must be expressed in a comma delimited list format.</p> <p>Units None</p>
StopCondListN / Parameter	<p>The StopCondListN option allows the user to enter all the parameters used for the propagator stopping condition. See the StopCondListN/Condition Option/Field for additional details to the StopCondListN option.</p> <p>Default DefaultSC.ElapsedSecs</p> <p>Limits Any single element user accessible spacecraft parameter followed by an equal sign</p> <p>Units None</p>
StopCondListN / Condition	<p>The StopCondListN option allows the user to enter the propagator stopping condition's value for the StopCondListN Parameter field.</p> <p>Default 8640.0</p> <p>Limits Real Number, Array element, Variable, spacecraft parameter, or any user defined parameter</p> <p>Units Dependant on the condition selected.</p>

Examples

Script Syntax

```
Propagate Mode BackProp PropagatorName(SatList1,fStopCondList1g) ...
BackPropPropagatorName (SatListN, {StopCondList})
```

Script Examples

```
% Single spacecraft propagation with one stopping condition
% Syntax #1
Propagate DefaultProp(DefaultSC, {DefaultSC.ElapsedSecs = 8640.0});

% Single spacecraft propagation with one stopping condition
% Syntax #2
Propagate DefaultProp(DefaultSC) {DefaultSC.ElapsedSecs = 8640.0};

% Single spacecraft propagation by one integration step
Propagate DefaultProp(DefaultSC);
```



```
% Multiple spacecraft propagation by one integration step
Propagate DefaultProp(Sat1, Sat2, Sat3);

% Single formation propagation by one integration step
Propagate DefaultProp(DefaultFormation);

% Single spacecraft backwards propagation by one integration step
Propagate Backwards DefaultProp(DefaultSC);

% Two spacecraft synchronized propagation with one stopping condition
Propagate Synchronized DefaultProp(Sat1, Sat2, {DefaultSC.ElapsedSecs = 8640.0})

% Multiple spacecraft propagation with multiple stopping conditions and propagation
% Syntax #1
Propagate Prop1(Sat1,Sat2, {Sat1.ElapsedSecs = 8640.0, Sat2.MA = 90}) ...
Prop2(Sat3, {Sat3.TA = 0.0});

% Multiple spacecraft propagation with multiple stopping conditions and propagation
% Syntax #2
Propagate Prop1(Sat1,Sat2) {Sat1.ElapsedSecs = 8640.0, Sat2.MA = 90} ...
Prop2(Sat3), {Sat3.TA = 0.0};
```

Report

Report — Output a report

Synopsis

Description

The report command allows the user find parameters of the orbit and the spacecraft at particular moments in time. This command is inserted into the mission tree at various locations in the mission tree. The parameters found by this command are placed into a report file that can be accessed at a later time.

Options

ReportName	The ReportName option allows the user to specift the ReportFile for data output.
Default	None
Limits	Any ReportFile created
Units	None
DataList	The DataList option allows the user to output data to the Filename specified by the ReportName. Multiple objects can be in the DataList when they are separated by spaces.
Default	None
Limits	Spacecraft parameter, Array, Variable, String, or any other single user defined parameter
Units	None

Examples

Script Syntax

```
Report ReportName DataList
```

Script Examples

```
%Report the time and position of DefaultSC  
Report DefaultReport DefaultSC.AlModJulian DefaultSC.X DefaultSC.Y DefaultSC.Z;
```

Save

Save — Under construction.

Synopsis

Under construction.

Under construction.

Under construction.

Examples

Script Syntax

Under construction.

Script Examples

Under construction.

ScriptEvent

ScriptEvent — Perform a ScriptEvent command

Synopsis

Overview

The ScriptEvent command allows a user to enter in script within the GUI in the mission sequence. This can be useful in a number of ways if its easier to enter the script than to use the GUI interface. Also, if multiple things needed to be changed rapidly, this could be useful. Additionally, the ScriptEvent allows a user to have more freedom if the GUI is problematic and won't allow the user to perform the desired operation.

Options

<Statements>	Default	None
	Limits	Any valid line of GMAT script
	Units	None

Examples

Script Syntax

```
BeginScript
    <Statements>
EndScript;
```

Script Examples

```
% Assignment Command inside Script Event

BeginScript
    GMAT testVar = 24;
EndScript;
```

Stop

Stop — Perform a stop command

Synopsis

Description

The Stop command simply ends a mission in GMAT. Whether placed in the script or GUI, Stop will halt the execution of the mission and have GMAT report back that the Command Sequence was intentionally interrupted.

Examples

Script Syntax

```
Stop
```

Script Examples

```
% Stop between propagation sequences
Propagate DefaultProp(DefaultSC) DefaultSC.ElapsedSecs = 8640.0;
Stop;
Propagate DefaultProp(DefaultSC) DefaultSC.ElapsedDays = 10.0;
```

Target

Target — Perform a targeting sequence

Synopsis

```
Target SolverName(options)
  Vary SolverName(Vary arguments);
  Achieve SolverName(Achieve arguments);
  [Other Commands;]
  ...
EndTarget
```

Description

When building a mission, the targeting sequence is one of the most useful tools you have at your disposal. It allows you to use the power of a differential corrector to find solutions to complex two and three-body problems with a limited number of known values and several variables. Target is the base command that is appended with more commands such as Vary, Maneuver, Propagate, and Achieve in order to meet your goals. This makes it the first part of the sequence to be put into the Mission tree.

Arguments

SolverName The SolverName option allows the user to choose between any previously created differential correctors for use in a targeting sequence. For example, to begin a targeting sequence using DefaultDC, the script is Target DefaultDC.

Type DifferentialCorrector

Options

SolveMode Tells the targeter state machine how to manage the targeter loop. When running in the RunInitialGuess mode, the target loop is executed one time using the initial values of the targeter variables. No targeting is performed. When running in Solve mode, the targeter searches for a solution satisfying the targeter goals.

Type Enumeration

Values

- Solve (default)
- RunInitialGuess

ExitMode Tells the Mission Control Sequence how to proceed after the targeter loop has finished running. When running in DiscardAndContinue mode, subsequent calls to the same targeter loop will reset the variables to their initial values. If running in SaveAndContinue mode, the targeter variables used on a subsequent run start with the values obtained the last time the targeter loop was run. In Stop mode, the mission run halts when the targeter loop completes its work.

Type Enumeration

Values

- DiscardAndContinue (default)

- SaveAndContinue
- Stop

Examples

Example 11. Targeting geosynchronous orbit using an impulsive burn

```
Target DefaultDC
  Vary DefaultDC(DefaultIB.Element1 = 0.5);
  Maneuver DefaultIB(DefaultSC);
  Propagate DefaultProp(DefaultSC) {DefaultSC.Earth.Apoapsis};
  Achieve DefaultDC(DefaultSC.Earth.RMAG = 42165.0);
EndTarget
```

Toggle

Toggle — Perform a toggle command

Synopsis

Description

The Toggle command is useful in turning on/off certain plots so that they do not become unintelligible. They are particularly useful if a mission requires multiple sequences to fine-tune the answer. If this is the case, it might be better to turn off a plot until the final or second-to-last sequence so that the rougher sequences don't obstruct the view of what is going on in the finesse sequences.

Options

OutputNames	The Toggle option allows the user to assign the Plot/Report(s) to be toggled. When more than one Plot/Report is being toggled they need to be separated by a space.
	Default DefaultOpenGL
	Limits Any OpenGL, Report, XYplot, or any other Plot/Report type
	Units None
Arg	The Arg option allows the user to turn off or on the data output to a Plot/Report.
	Default On
	Limits On or Off
	Units None

Exmaples

Script Syntax

```
Toggle OutputNames Arg
```

Script Examples

```
% Turn off Report file for the first day of propagation
Toggle ReportFile1 Off
Propagate DefaultProp(DefaultSC, {DefaultSC.ElapsedDays = 1});
Toggle ReportFile1 On
Propagate DefaultProp(DefaultSC, {DefaultSC.ElapsedDays = 1});

% Turn off XYPlot and Report file for the first day of propagation
Toggle XYPlot1 ReportFile1 Off
Propagate DefaultProp(DefaultSC, {DefaultSC.ElapsedDays = 1});

Toggle XYPlot1 ReportFile1 On
```



```
Propagate DefaultProp(DefaultSC, {DefaultSC.ElapsedDays = 1})
```

Vary

Vary — Perform a vary command

Synopsis

Description

The Vary command is used in conjunction with the Target and Optimize Commands. The Vary command varies a particular parameter within the Target and Optimize Commands until certain conditions are met. It is a highly useful command for creating missions. Multiple Vary commands can be used within the same targeting or optimizing sequence.

Options

Parameters Associated with All Solvers

SolverName	The SolverName option allows the user to choose which solver to assign to the vary command.	
	Default	DefaultDC
	Limits	Any user defined solver
	Units	None
Variable	The Variable option allows the user to select any single element user defined parameter, except a number, to vary. For example, DefaultIB.V, DefaultIB.N, DefaultIB.Element1, DefaultSC.TA, Array(1,1), and Variable are all valid values. The three element burn vector or multidimensional Arrays are not valid values.	
	Default	DefaultIB.V
	Limits	Spacecraft parameter, Array element, Variable, or any other single element user defined parameter, excluding numbers
	Units	None
InitialGuess	The InitialGuess option allows the user to set the initial guess for the selected Variable.	
	Default	0.5
	Limits	Real Number, Array element, Variable, or any user defined parameter that obeys the conditions for the selected Variable object
	Units	km/s
Lower	The Lower option allows the user to set Arg3 to the lower bound of the quantity being varied.	
	Default	0.0
	Limits	Real Number, Array element, Variable, or any user defined parameter (Upper > Lower)

	Units	None
Upper	The Upper option allows the user to set Arg4 to the upper bound of the quantity being varied.	
	Default	3.14159
	Limits	Real Number, Array element, Variable, or any user defined parameter (Upper > Lower)
	Units	None

Parameters Associated with Differential Corrector

Perturbation	The Perturbation option is set by specifying a value for Arg1. The value of Arg1 is the perturbation size in calculating the finite difference derivative.	
	Default	1e-4
	Limits	Real Number, Array element, Variable, or any user defined parameter > 0
	Units	None
MaxStep	The MaxStep option is set by specifying a value for Arg2. The value of Arg2 limits the size of the step taken during an interaction of the differential corrector.	
	Default	0.2
	Limits	Real Number, Array element, Variable, or any user defined parameter > 0
	Units	None

Parameters Associated with fmincon Optimizer

Additive Scale Factor	The AdditiveScaleFactor Field is used to nondimensionalize the independent variable. fmincon sees only the nondimensional form of the variable. The nondimensionalization is performed using the following equation: $x_n = (x_d - a)/m$. (x_n is the non-dimensional parameter. x_d is the dimensional parameter. a = additive scale factor. m = multiplicative scale factor.)	
	Default	0
	Limits	Real Number, Array element, Variable, or any user defined parameter
	Units	None
Multiplicative Scale Factor	The MultiplicativeScaleFactor Field is used to nondimensionalize the independent variable. fmincon sees only the nondimensional form of the variable. The nondimensionalization is performed using the following equation: $x_n = (x_d - a)/m$. (x_n is the non-dimensional parameter. x_d is the dimensional parameter. a = additive scale factor. m = multiplicative scale factor.)	
	Default	1.0

Limits	Real Number, Array element, Variable, or any user defined parameter
Units	None

Examples

Script Syntax

```
Vary SolverName(Variable=InitialGuess,{Perturbation=Arg1, MaxStep=Arg2,  
Lower=Arg3,...Upper=Arg4, AdditiveScalefactor=Arg6,MultiplicativeScalefactor=
```

Script Examples

```
% Impulsive Burn Vary Command  
  
Vary DefaultDC(DefaultIB.V = 0.5, {Perturbation = 0.0001, MaxStep = 0.2,  
Lower = 0, Upper = 3.14159});
```

While

While — Run a while loop

Synopsis

Description

The while loop is a control logic function that allows GMAT to check the spacecraft's status on a given parameter while performing a command or another control logic function within the mission sequence. When a spacecraft has reached the given property, the while loop will check its condition and react according to the equation defined in the loop's dialog box.

Options

<logical expression>	Arg1 and Arg2 can be any of the following: Real Number, Array, Variable, Spacecraft Parameter or any other user defined parameter.	
	Default	DefaultSC.ElapsedDays < 1.0
	Limits	Arg1 < Arg2 and < can be > , < , >= , <= , == , ~=
<Statements>	Units	None
	Default	None
	Limits	Any script line that can be in the mission sequence
	Units	None
	The option allows the user to set an OR operator in between s.	
	Default	None
&	Limits	None
	Units	None
	The & option allows the user to set an AND operator in between <logical expression>s.	
	Default	None
	Limits	None
	Units	None

Examples

Script Syntax

- Simple While Loop

```
While <logical expression>;
```

```
    <Statements>;  
EndWhile;
```

- Compound While Loop

```
While <logical expression> | <logical expression> & <logical expression>;  
    <Statements>  
EndWhile;
```

Script Examples

```
While DefaultSC.ElapsedDays < 1;  
Propagate DefaultProp (DefaultSC , DefaultSC.Elapsed Days = 0.01);  
EndWhile;  
  
While MyVariable < MyArray(1,1);  
MyArray(1,1) = 5;  
EndWhile;
```

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