

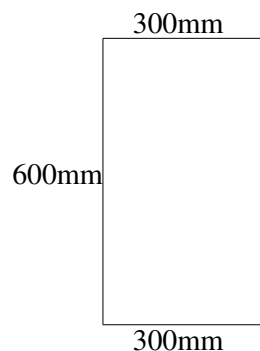
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ai24btech11028 - Ronit Ranjan

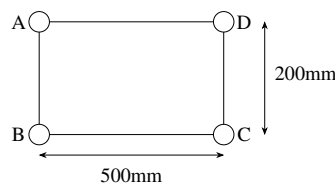
- 1) The critical Mach number for a flat plate of zero thickness, at zero angle of attack, is _____
- 2) A damped single degree-of-freedom system is vibrating under a harmonic excitation with an amplitude ratio of 2.5 at resonance. The damping ratio of the system is _____
- 3) The cross-section of a long thin-walled member is as shown in the figure. When subjected to pure twist, point A



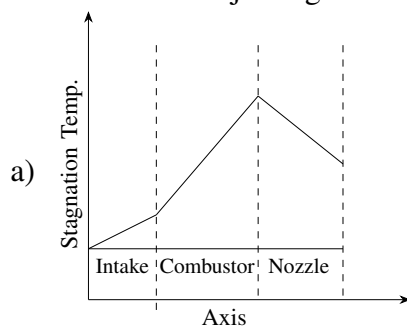
- a) does not move horizontally or axially, but moves vertically
 - b) does not move axially, but moves both vertically and horizontally
 - c) does not move horizontally, vertically or axially
 - d) does not move vertically or axially, but moves horizontally
- 4) The channel section of uniform thickness 2mm shown in the figure is subjected to a torque of 10 Nm. If it is made of a material with shear modulus of 25 GPa, the twist per unit length in radians/m is _____

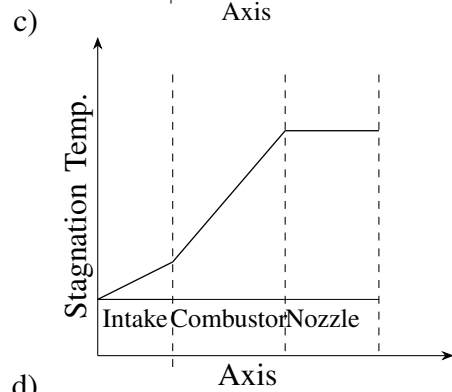
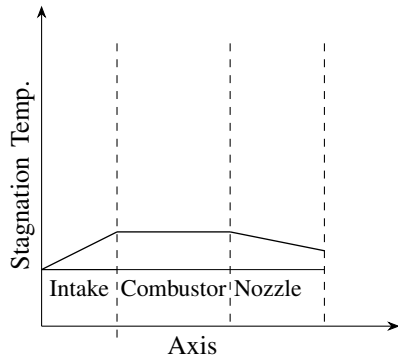
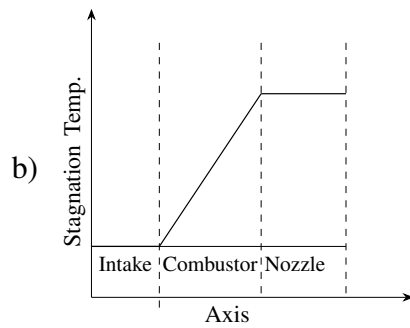


- 5) The stiffened cross-section of a long slender uniform structural member is idealized as shown in the figure below. The lumped areas at A, B, C and D have equal cross-sectional area of 3 cm². The webs AB, BC, CD and DA are each 5 mm thick. The structural member is subjected to a twisting moment of 10 kNm. The magnitudes of the shear flow in the webs, q_{AB} , q_{BC} , q_{CD} , and q_{DA} in kN/m are, respectively



- a) 20, 20, 20, 20
 b) 0, 0, 50, 50
 c) 40, 40, 0, 0
 d) 50, 50, 50, 50
- 6) Consider two engines P and Q. In P, the high pressure turbine blades are cooled with a bleed of 5% from the compressor after the compression process and in Q the turbine blades are not cooled. Comparing engine P with engine Q, which one of the following is NOT TRUE?
- a) Turbine inlet temperature is higher for engine P
 b) Specific thrust is higher for engine P
 c) Compressor work is the same for both P and Q
 d) Fuel flow rate is lower for engine P
- 7) The mass flow rate of air through an aircraft engine is 10 kg/s. The compressor outlet temperature is 400 K and the turbine inlet temperature is 1800 K. The heating value of the fuel is 42 MJ/kg and the specific heat at constant pressure is 1 kJ/kg-K. The mass flow rate of the fuel in kg/s is approximately _____
- 8) For a given inlet condition, if the turbine inlet temperature is fixed, what value of compressor efficiency given below leads to the lowest amount of fuel added in the combustor of a gas turbine engine?
- a) 1
 b) 0.95
 c) 0.85
 d) 0.8
- 9) A gas turbine engine is mounted on an aircraft which can attain a maximum altitude of 11 km from sea level. The combustor volume of this engine is decided based on conditions at
- a) sea level
 b) 8 km altitude
 c) 5.5 km altitude
 d) 11 km altitude
- 10) Consider the low earth orbit (LEO) and the geo synchronous orbit (GSO). Then
- a) ΔV requirement for launch to LEO is greater than that for GSO, and altitude of LEO is lower than that of GSO
 b) ΔV requirement for launch to LEO is lower than that for GSO, and altitude of LEO is lower than that of GSO
 c) ΔV requirement for launch to LEO is greater than that for GSO, and altitude of LEO is greater than that of GSO
 d) ΔV requirement for launch to LEO is lower than that for GSO, and altitude of LEO is greater than that of GSO
- 11) Which one of the following shows the CORRECT variation of stagnation temperature along the axis of an ideal ram jet engine?





- 12) A rocket motor has a chamber pressure of 100 bar and chamber temperature of 3000 K. The ambient pressure is 1 bar. Assume that the specific heat at constant pressure is 1 kJ/kg-K. Also assume that the flow in the nozzle is isentropic and optimally expanded. The exit static temperature in K is
- 805
 - 845
 - 905
 - 945

A. Q.13 to Q.?? carry two marks each

- 13) $I = \iint_S (y^2 \hat{i} + z^2 \hat{j} + x^2 \hat{k}) \cdot (x \hat{i} + y \hat{j} + z \hat{k}) dS$, where S denotes the surface of the sphere of unit radius centered at the origin. Here \hat{i} , \hat{j} and \hat{k} denote three orthogonal unit vectors. The value of I is
