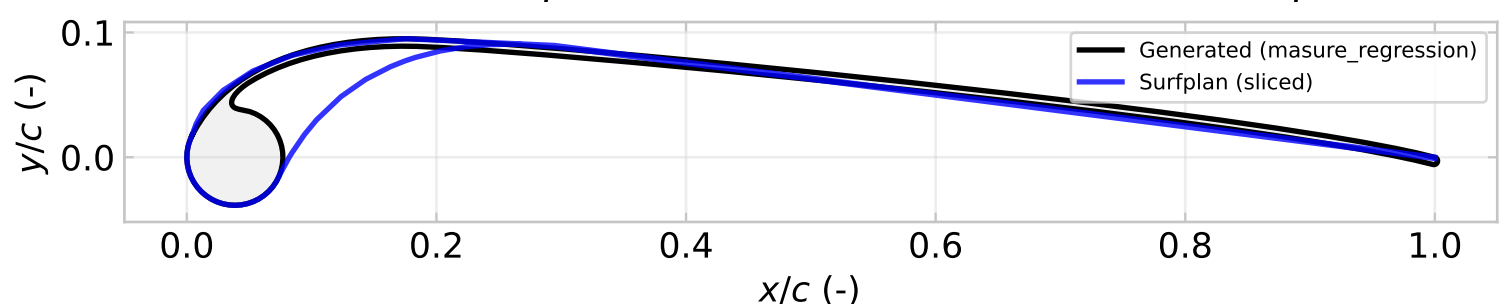
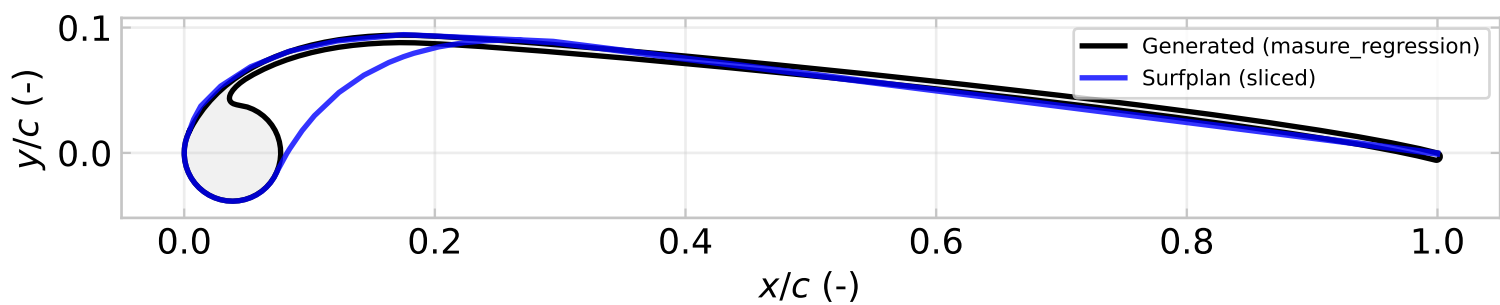
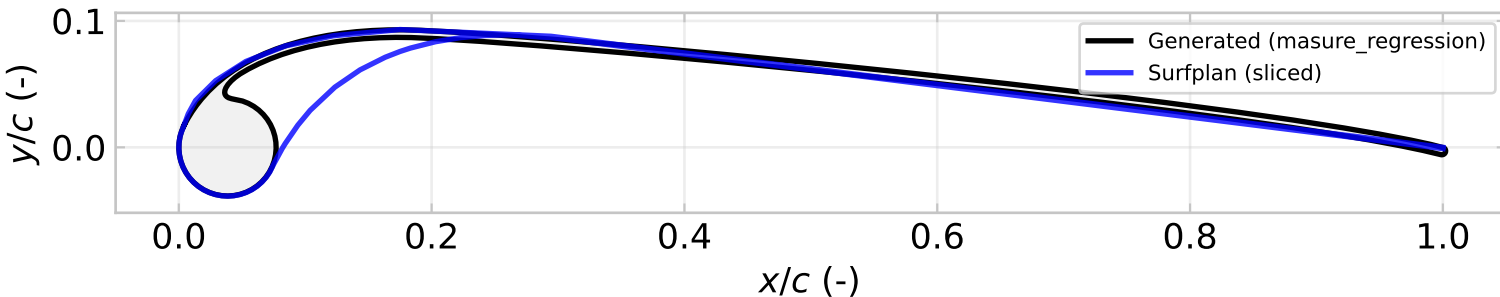
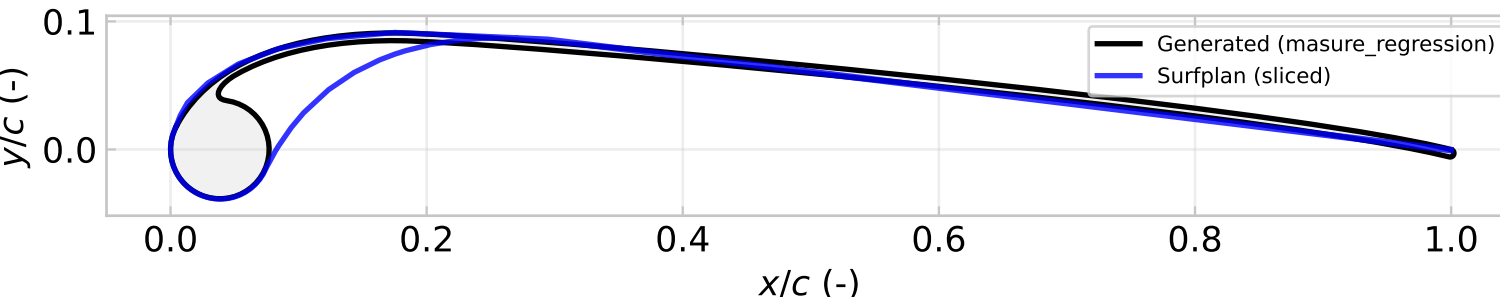
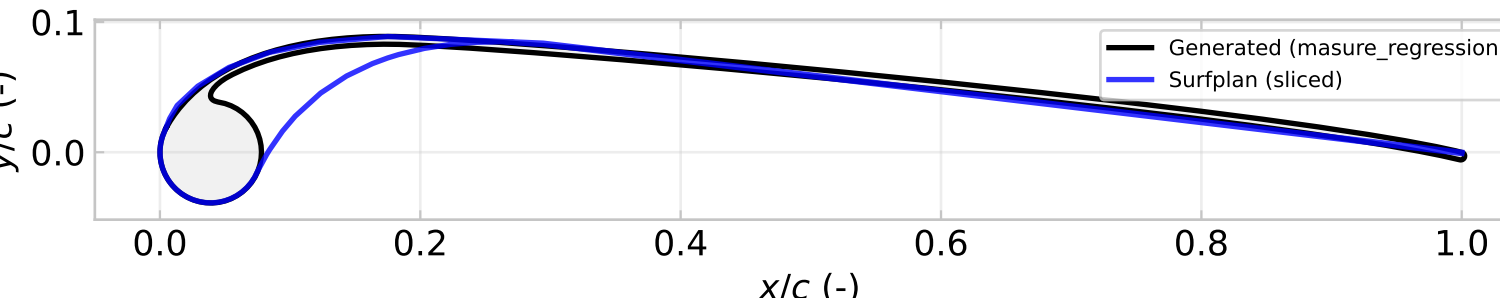
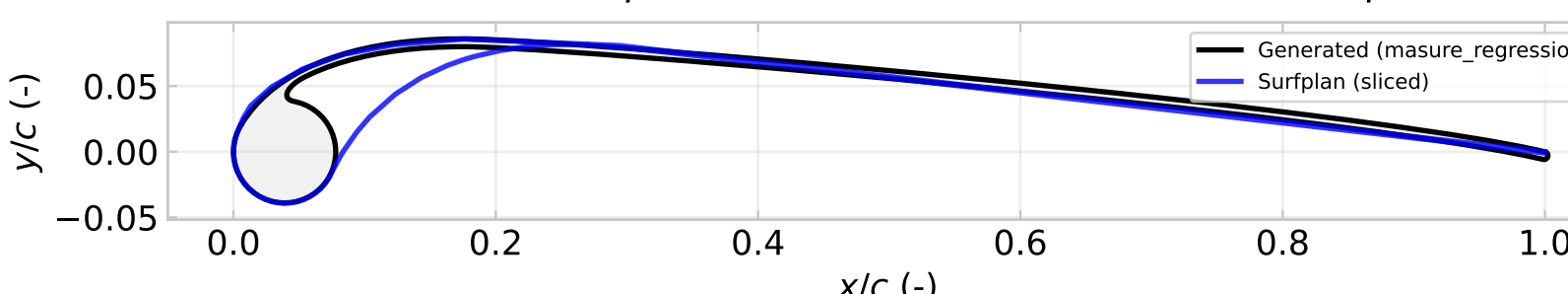
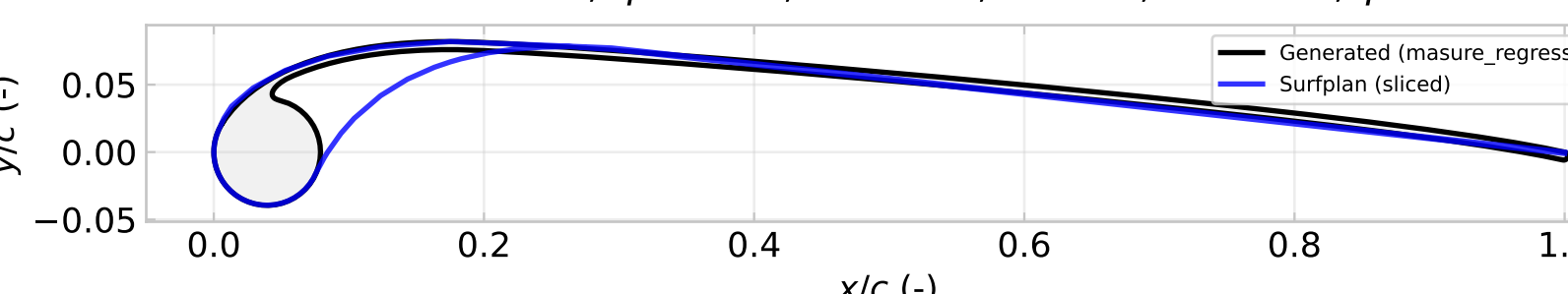
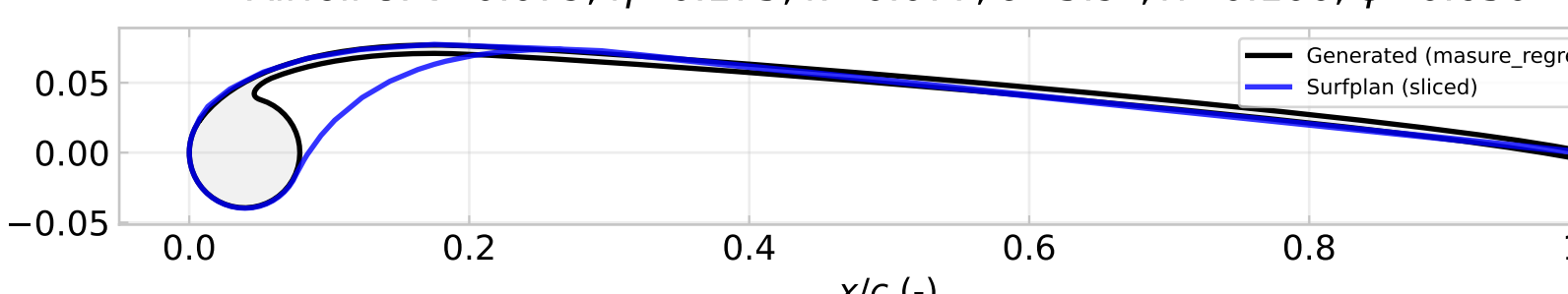
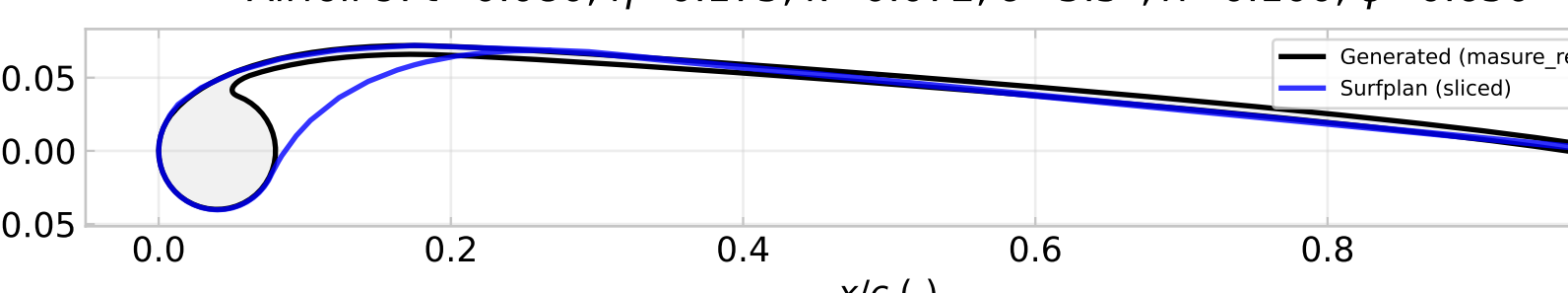
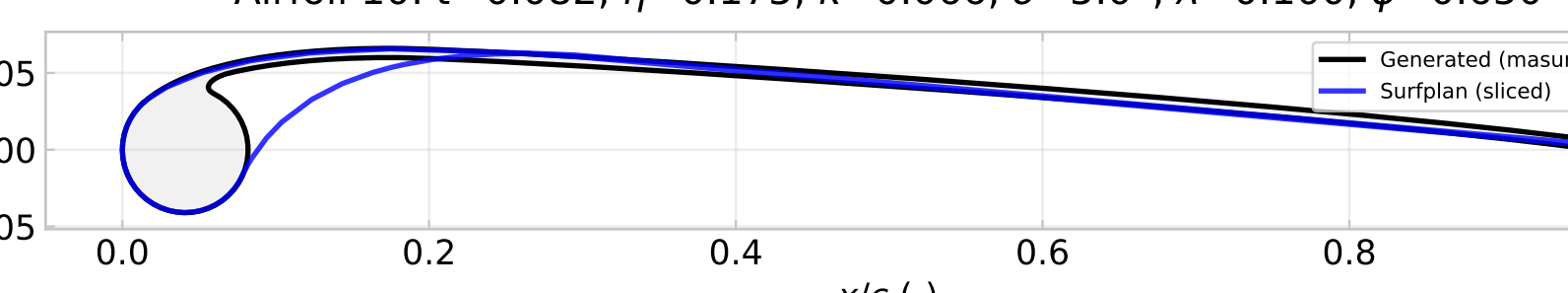
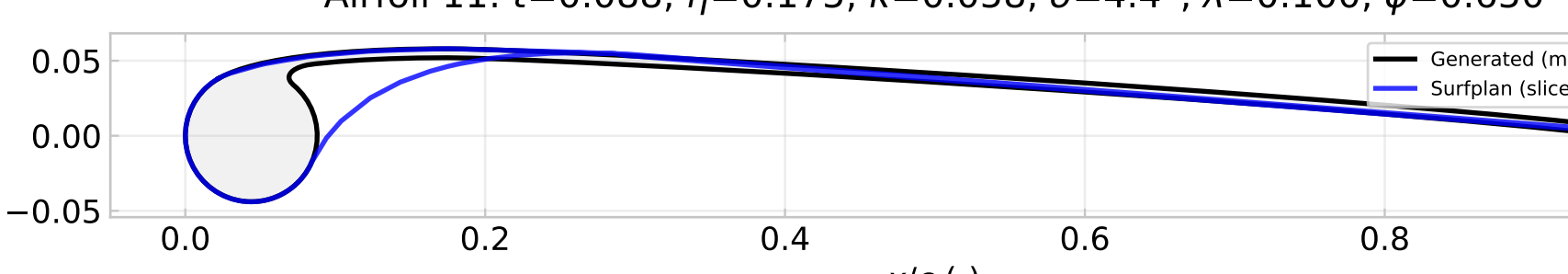
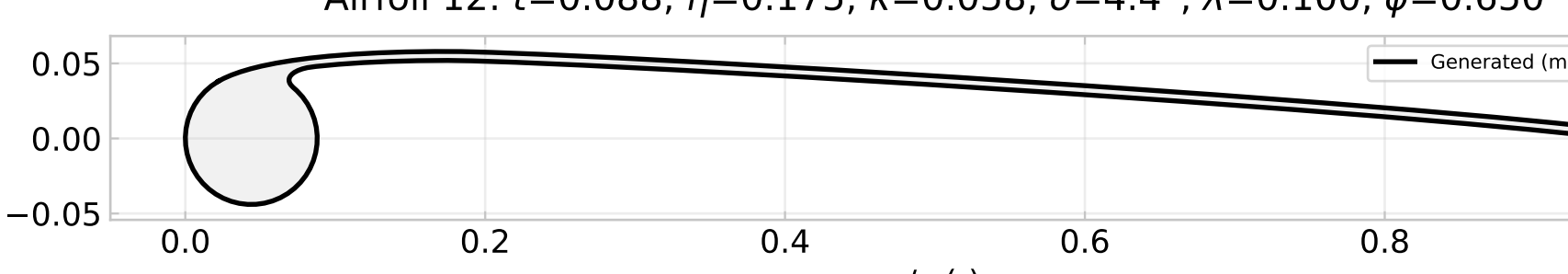
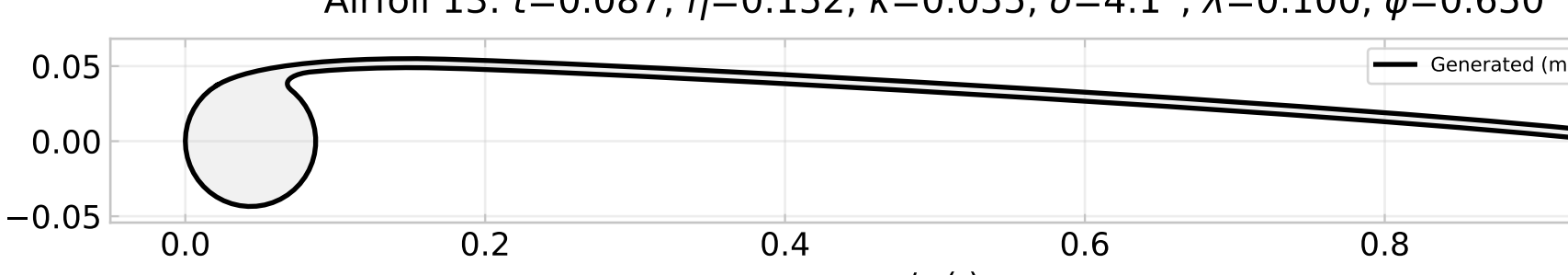
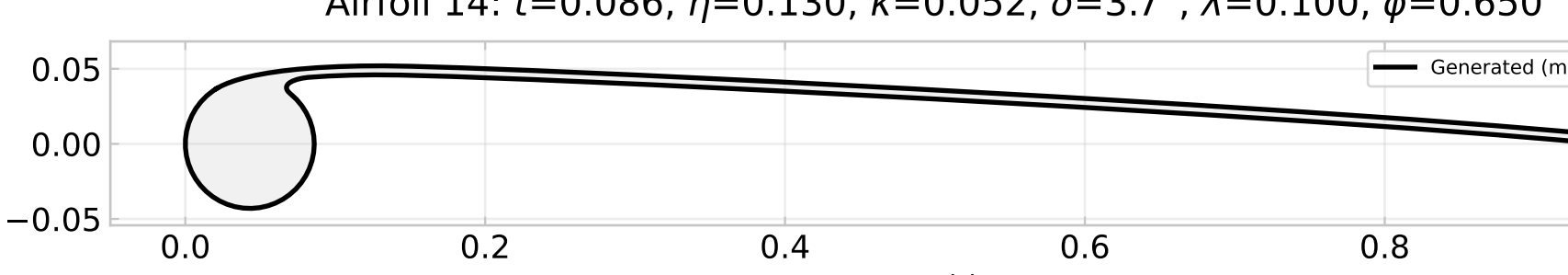
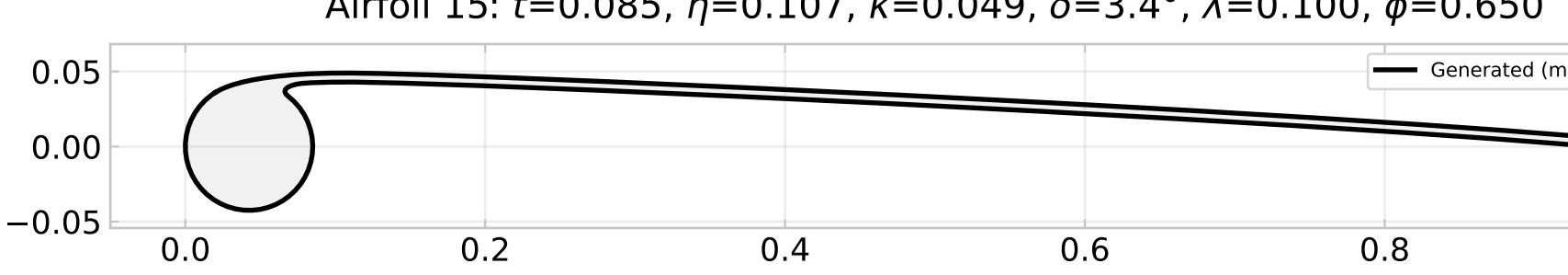
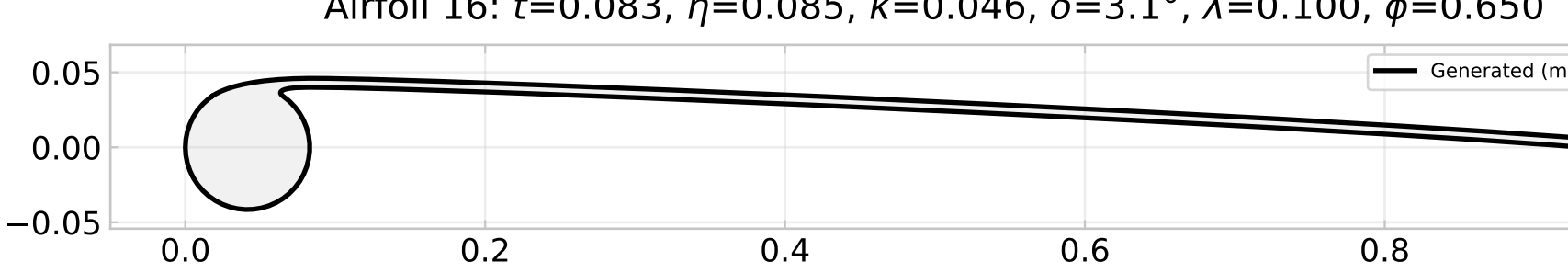
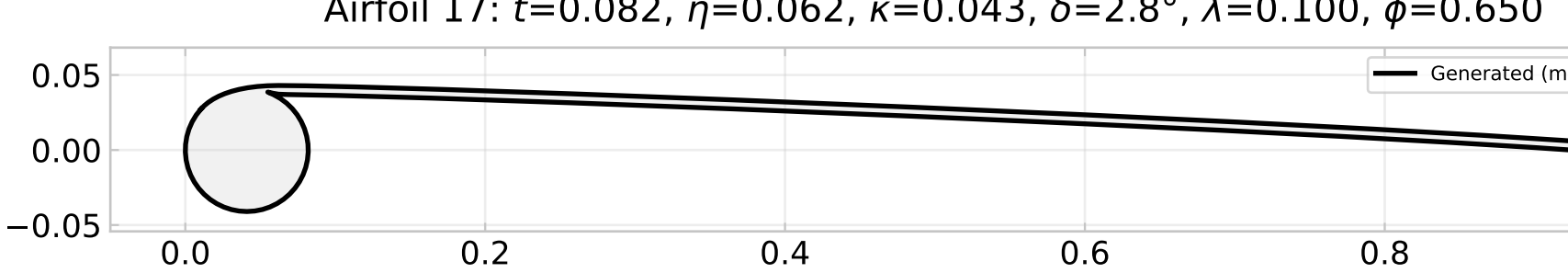


Airfoil 1:  $t=0.077$ ,  $\eta=0.175$ ,  $\kappa=0.095$ ,  $\delta=7.2^\circ$ ,  $\lambda=0.100$ ,  $\phi=0.650$ Airfoil 2:  $t=0.077$ ,  $\eta=0.175$ ,  $\kappa=0.094$ ,  $\delta=7.2^\circ$ ,  $\lambda=0.100$ ,  $\phi=0.650$ Airfoil 3:  $t=0.077$ ,  $\eta=0.175$ ,  $\kappa=0.093$ ,  $\delta=7.1^\circ$ ,  $\lambda=0.100$ ,  $\phi=0.650$ Airfoil 4:  $t=0.077$ ,  $\eta=0.175$ ,  $\kappa=0.091$ ,  $\delta=7.0^\circ$ ,  $\lambda=0.100$ ,  $\phi=0.650$ Airfoil 5:  $t=0.078$ ,  $\eta=0.175$ ,  $\kappa=0.089$ ,  $\delta=6.8^\circ$ ,  $\lambda=0.100$ ,  $\phi=0.650$ Airfoil 6:  $t=0.078$ ,  $\eta=0.175$ ,  $\kappa=0.086$ ,  $\delta=6.5^\circ$ ,  $\lambda=0.100$ ,  $\phi=0.650$ Airfoil 7:  $t=0.079$ ,  $\eta=0.175$ ,  $\kappa=0.082$ ,  $\delta=6.2^\circ$ ,  $\lambda=0.100$ ,  $\phi=0.650$ Airfoil 8:  $t=0.079$ ,  $\eta=0.175$ ,  $\kappa=0.077$ ,  $\delta=5.9^\circ$ ,  $\lambda=0.100$ ,  $\phi=0.650$ Airfoil 9:  $t=0.080$ ,  $\eta=0.175$ ,  $\kappa=0.072$ ,  $\delta=5.5^\circ$ ,  $\lambda=0.100$ ,  $\phi=0.650$ Airfoil 10:  $t=0.082$ ,  $\eta=0.175$ ,  $\kappa=0.066$ ,  $\delta=5.0^\circ$ ,  $\lambda=0.100$ ,  $\phi=0.650$ Airfoil 11:  $t=0.088$ ,  $\eta=0.175$ ,  $\kappa=0.058$ ,  $\delta=4.4^\circ$ ,  $\lambda=0.100$ ,  $\phi=0.650$ Airfoil 12:  $t=0.088$ ,  $\eta=0.175$ ,  $\kappa=0.058$ ,  $\delta=4.4^\circ$ ,  $\lambda=0.100$ ,  $\phi=0.650$ Airfoil 13:  $t=0.087$ ,  $\eta=0.152$ ,  $\kappa=0.055$ ,  $\delta=4.1^\circ$ ,  $\lambda=0.100$ ,  $\phi=0.650$ Airfoil 14:  $t=0.086$ ,  $\eta=0.130$ ,  $\kappa=0.052$ ,  $\delta=3.7^\circ$ ,  $\lambda=0.100$ ,  $\phi=0.650$ Airfoil 15:  $t=0.085$ ,  $\eta=0.107$ ,  $\kappa=0.049$ ,  $\delta=3.4^\circ$ ,  $\lambda=0.100$ ,  $\phi=0.650$ Airfoil 16:  $t=0.083$ ,  $\eta=0.085$ ,  $\kappa=0.046$ ,  $\delta=3.1^\circ$ ,  $\lambda=0.100$ ,  $\phi=0.650$ Airfoil 17:  $t=0.082$ ,  $\eta=0.062$ ,  $\kappa=0.043$ ,  $\delta=2.8^\circ$ ,  $\lambda=0.100$ ,  $\phi=0.650$ Airfoil 18:  $t=0.081$ ,  $\eta=0.040$ ,  $\kappa=0.041$ ,  $\delta=2.4^\circ$ ,  $\lambda=0.100$ ,  $\phi=0.650$ 