

aircraft

$\mathbf{x}_1 :-$

SFC, W_e, ESF

D

W_t, θ, ESF

$L, D, L/D$

L

W_s, W_f, θ

propulsion

$\mathbf{x}_2 : T$

aerodynamics

$\mathbf{x}_3 : \Lambda_{ht}, L_w, L_{ht}$

structures

$\mathbf{x}_4 : [t], [t_s], \lambda$

$t/c, AR_w, \Lambda_w, S_{ref}, S_{ht}, AR_{ht}$

