

# **Airfoil Analysis Report**

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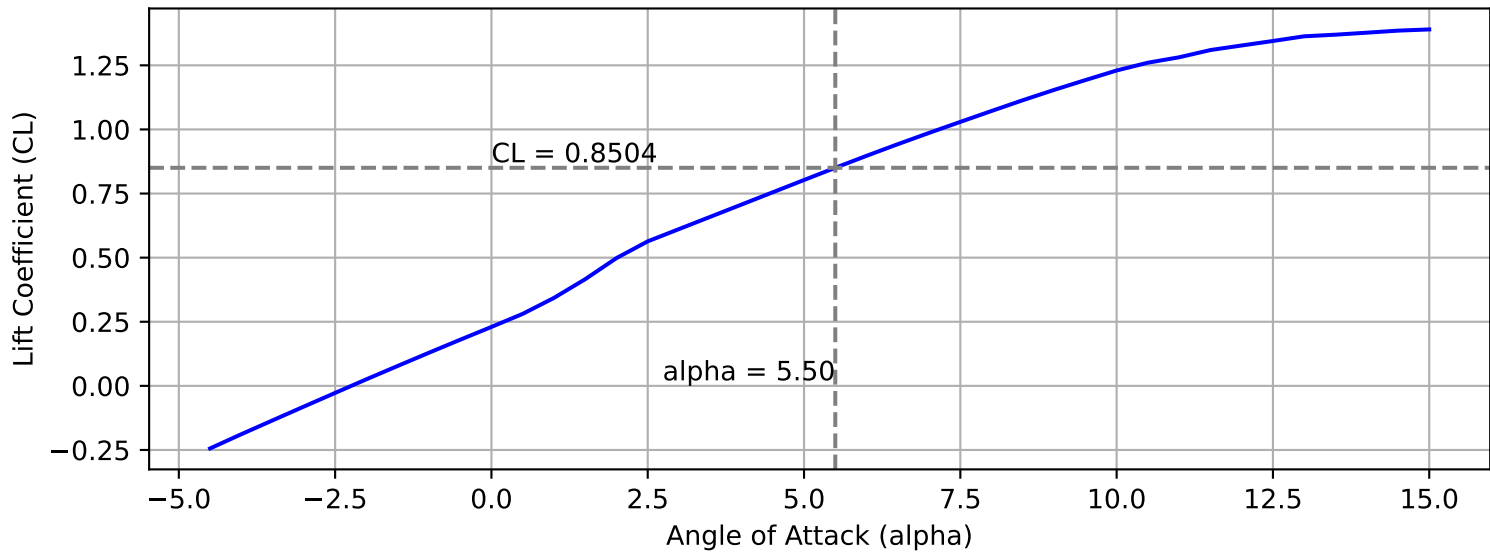
1. NACA 2412

2. NACA 2412

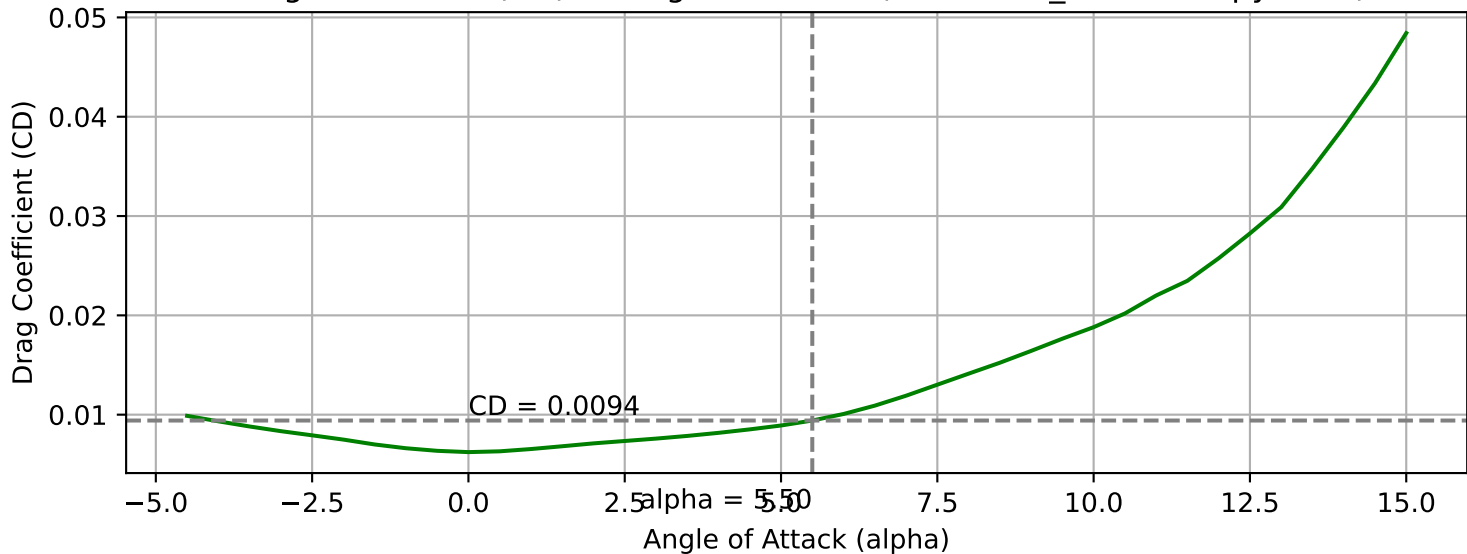
3. NACA 2412

# NACA 2412

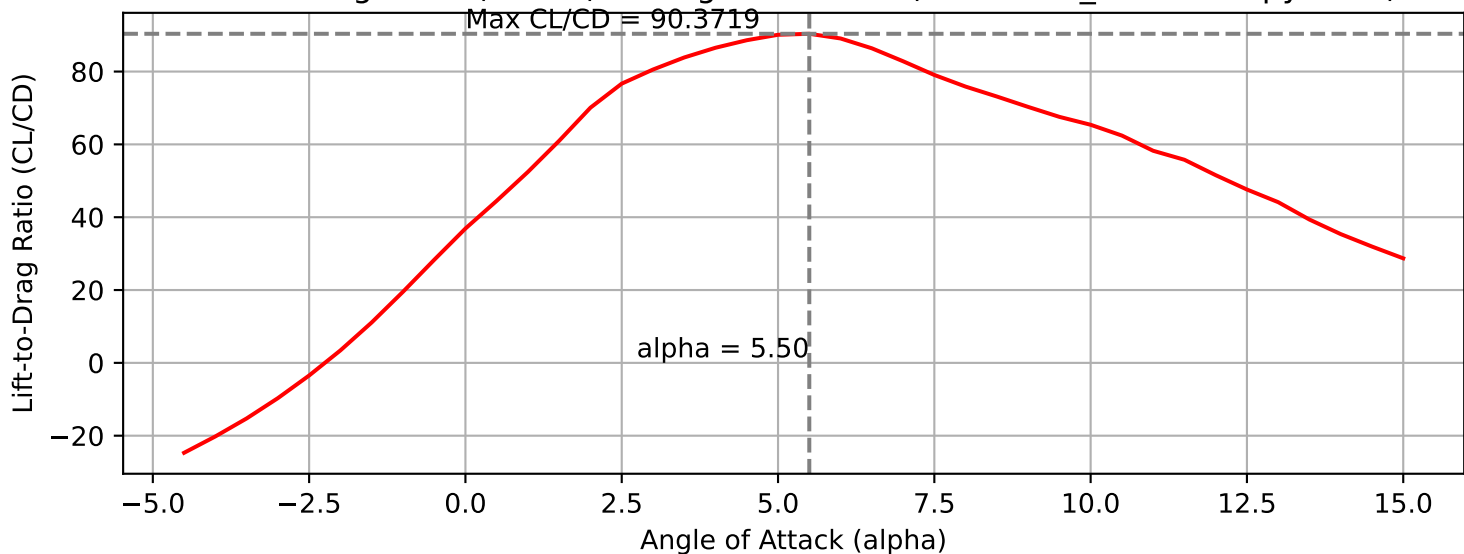
Lift Coefficient (CL) vs Angle of Attack (naca2412\_viscous copy 2.txt)



Drag Coefficient (CD) vs Angle of Attack (naca2412\_viscous copy 2.txt)



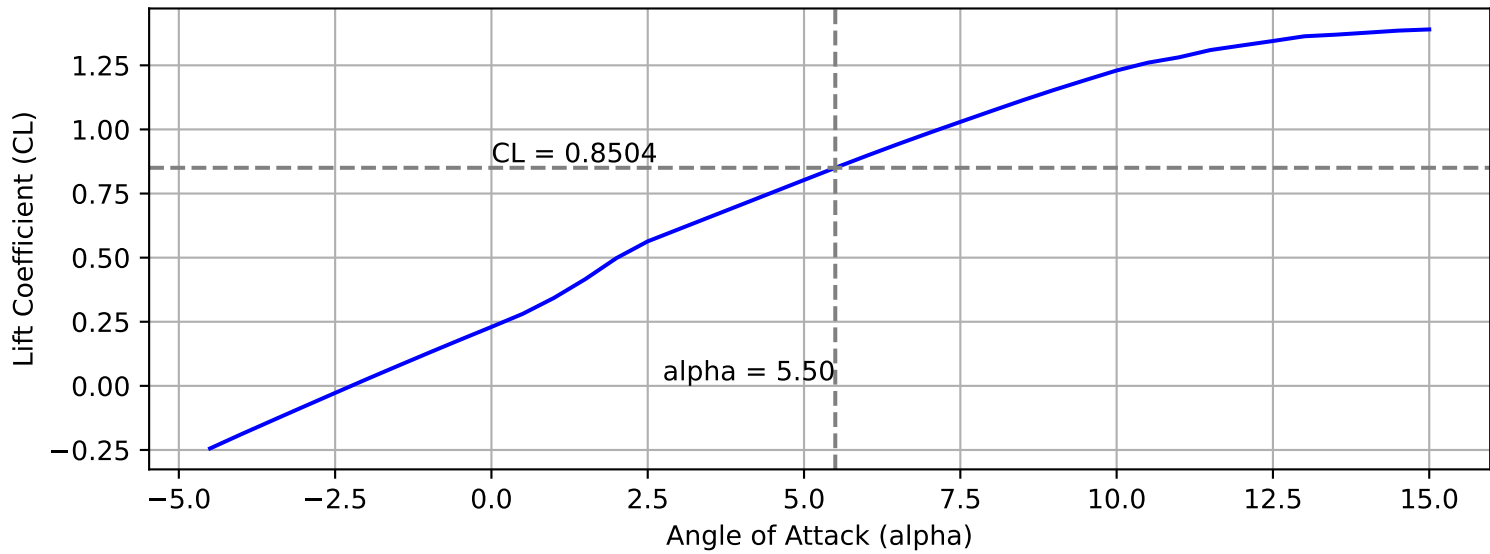
Lift-to-Drag Ratio (CL/CD) vs Angle of Attack (naca2412\_viscous copy 2.txt)



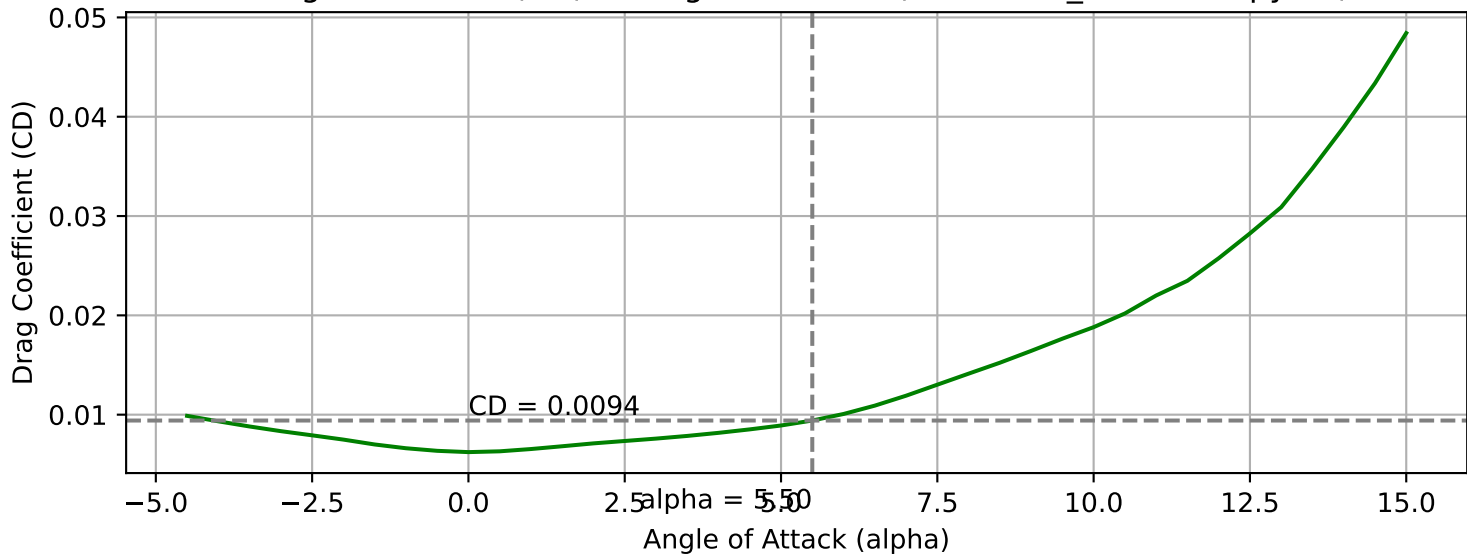
At alpha = 5.50: CL = 0.8504, CD = 0.0094, Max CL/CD = 90.3719

# NACA 2412

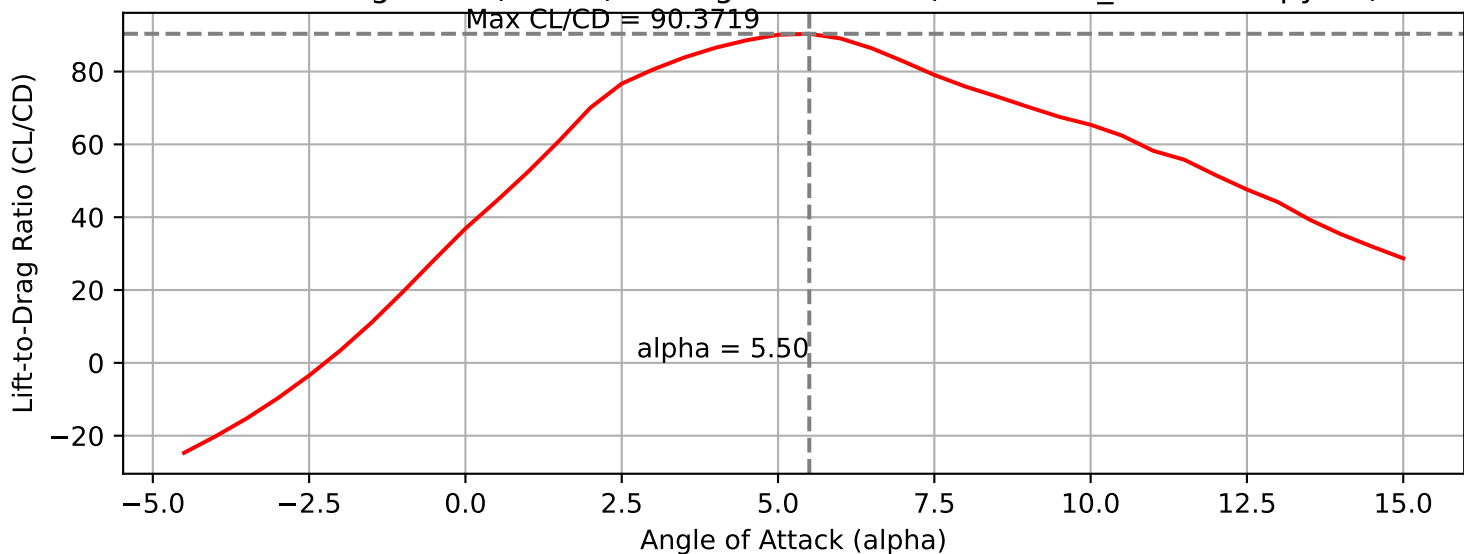
Lift Coefficient (CL) vs Angle of Attack (naca2412\_viscous copy.txt)



Drag Coefficient (CD) vs Angle of Attack (naca2412\_viscous copy.txt)



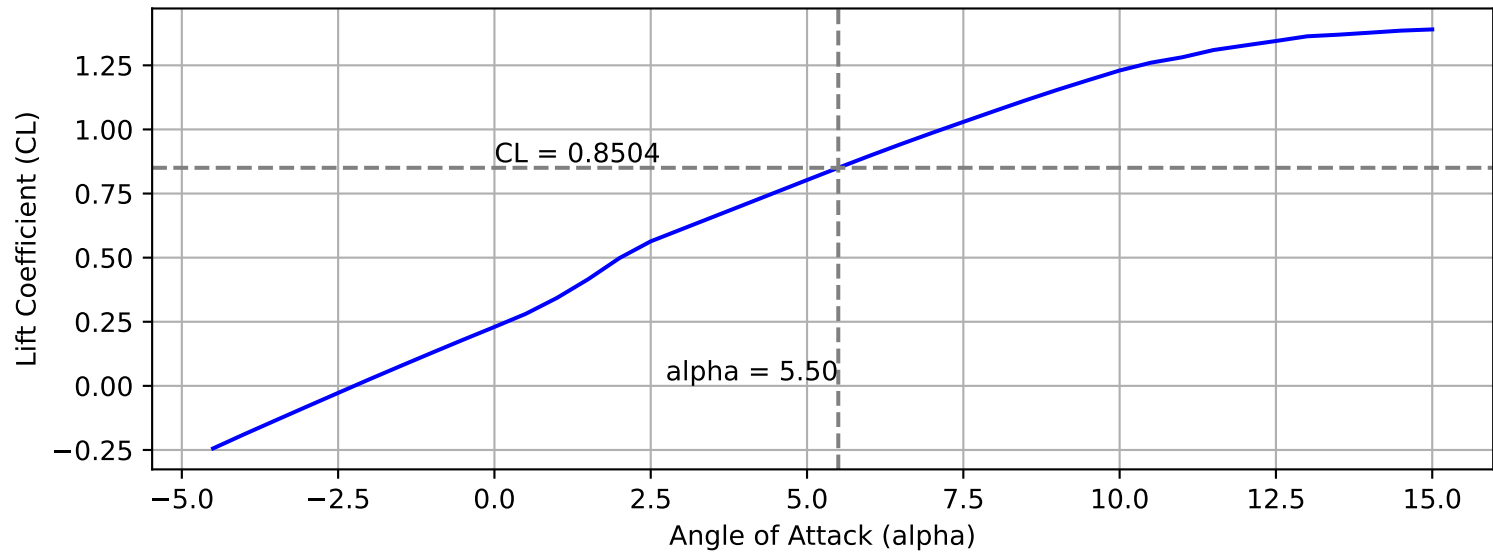
Lift-to-Drag Ratio (CL/CD) vs Angle of Attack (naca2412\_viscous copy.txt)



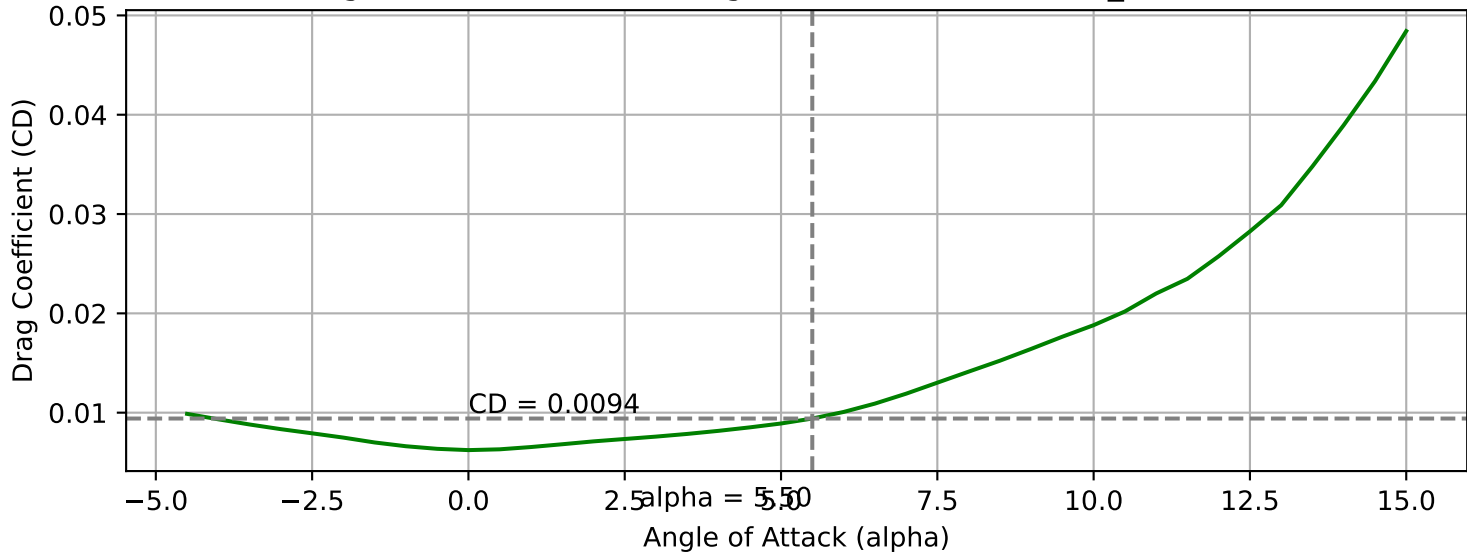
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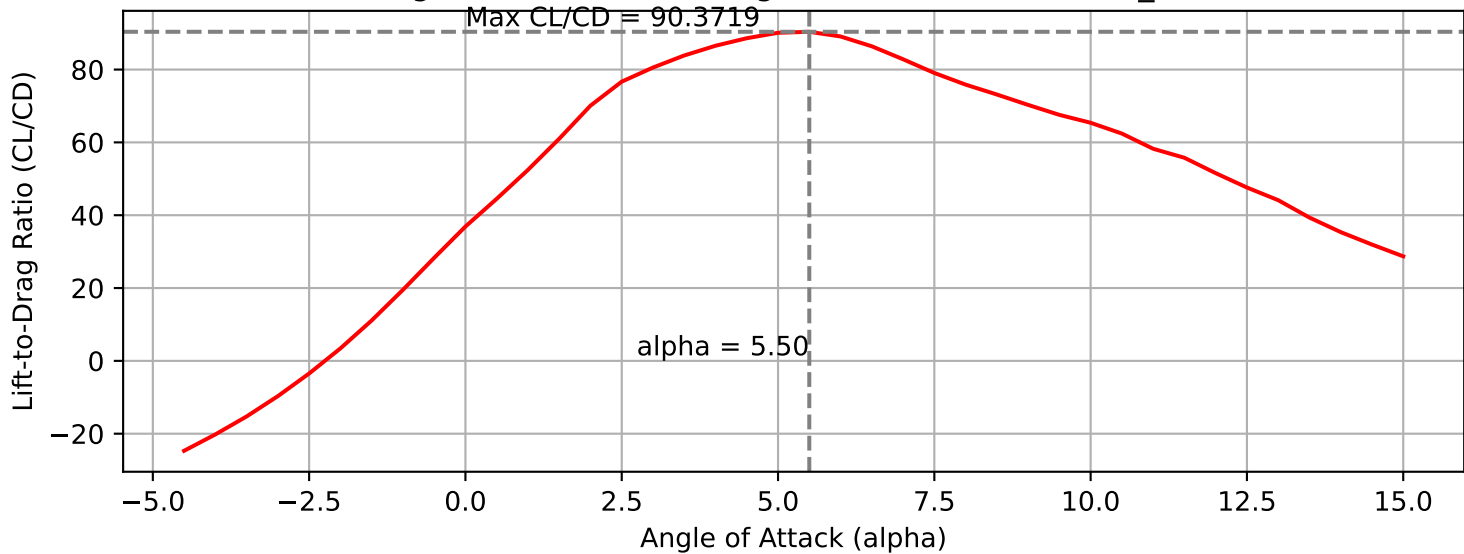
Lift Coefficient (CL) vs Angle of Attack (naca2412\_viscous.txt)



Drag Coefficient (CD) vs Angle of Attack (naca2412\_viscous.txt)



Lift-to-Drag Ratio (CL/CD) vs Angle of Attack (naca2412\_viscous.txt)



At alpha = 5.50: CL = 0.8504, CD = 0.0094, Max CL/CD = 90.3719