

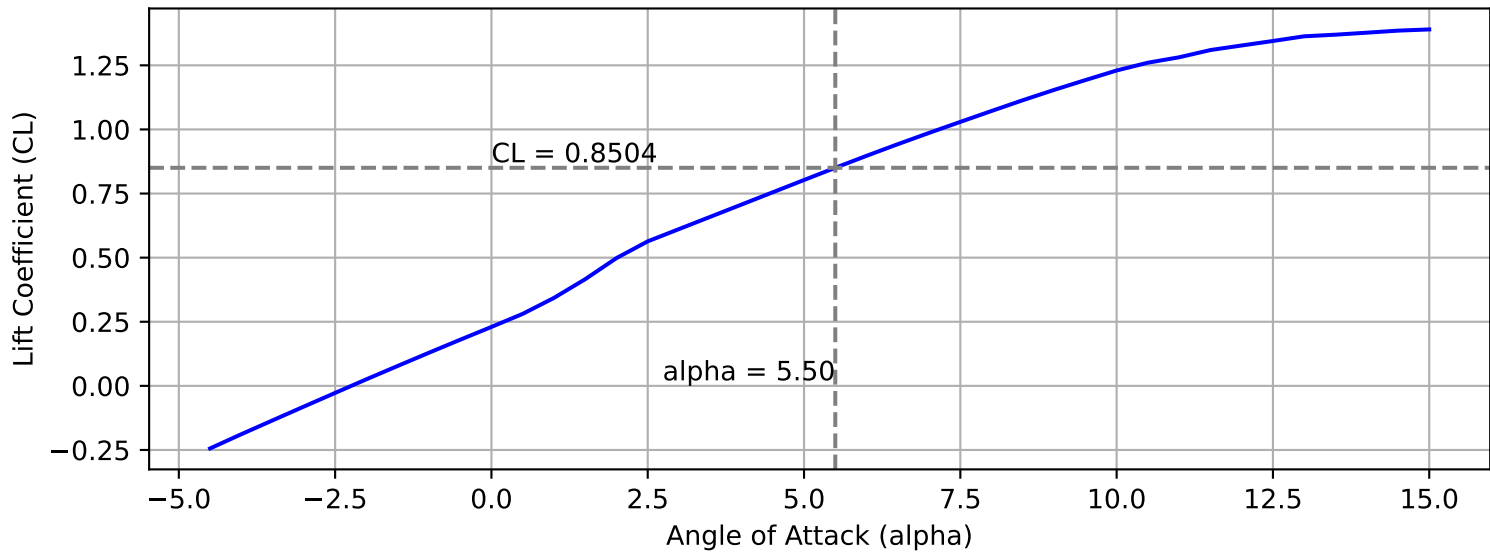
# **Airfoil Analysis Report**

# Table of Contents

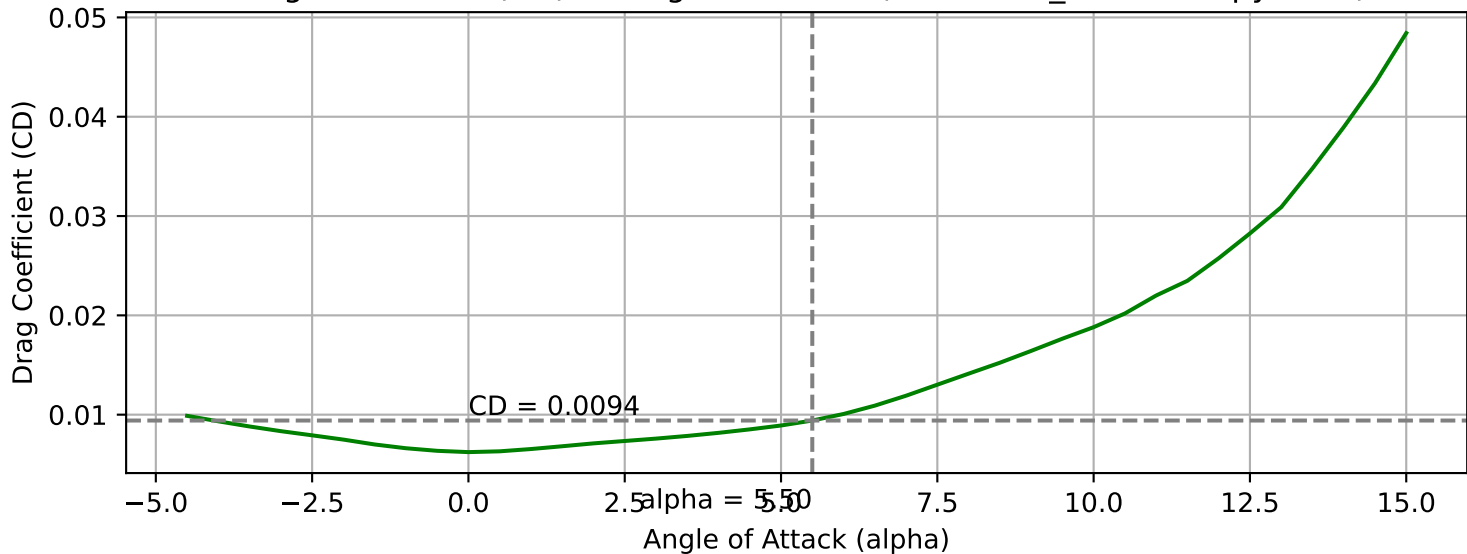
1. NACA 2412, CL at Max CL/CD: 0.8504	Page 1
2. NACA 2412, CL at Max CL/CD: 0.8504	Page 2
3. NACA 2412, CL at Max CL/CD: 0.8504	Page 3

# NACA 2412, $Re = 5.220e+05$

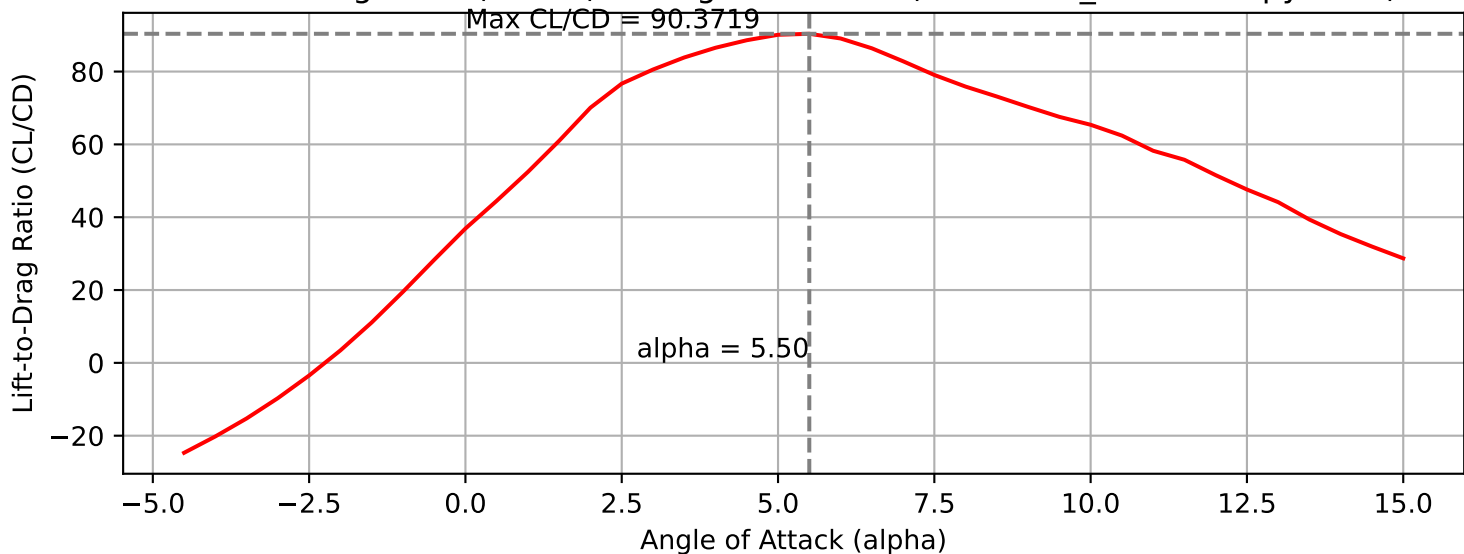
Lift Coefficient ( $C_L$ ) vs Angle of Attack (naca2412\_viscous copy 2.txt)



Drag Coefficient ( $C_D$ ) vs Angle of Attack (naca2412\_viscous copy 2.txt)



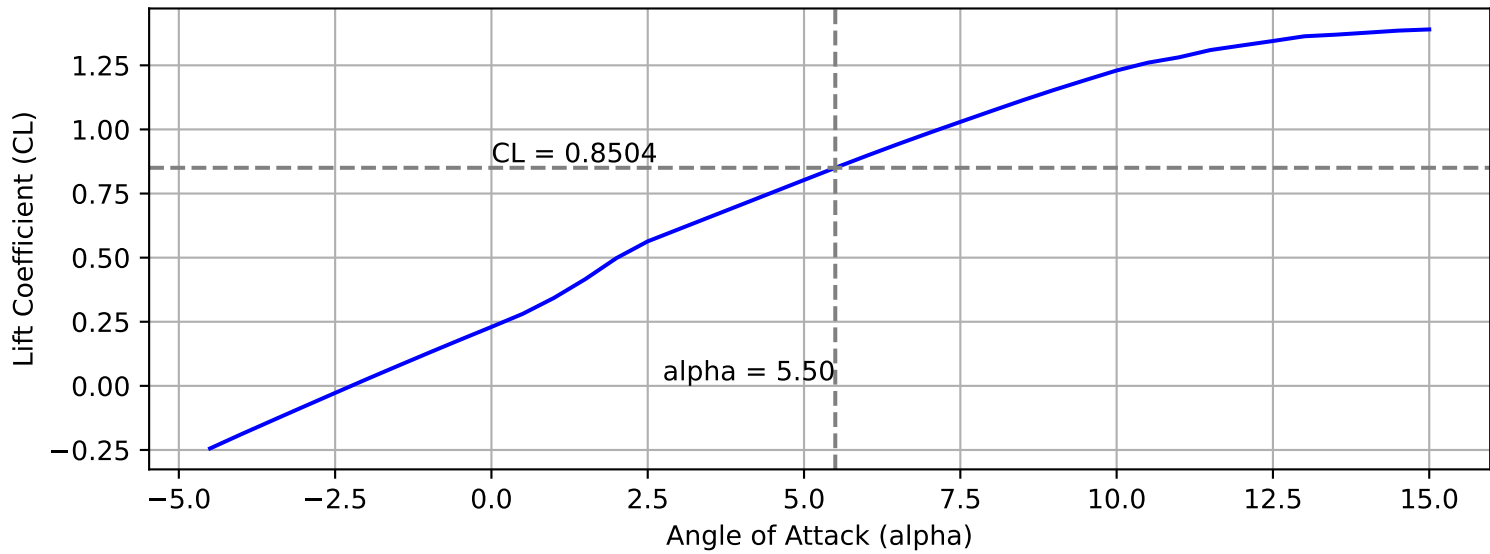
Lift-to-Drag Ratio ( $C_L/C_D$ ) vs Angle of Attack (naca2412\_viscous copy 2.txt)



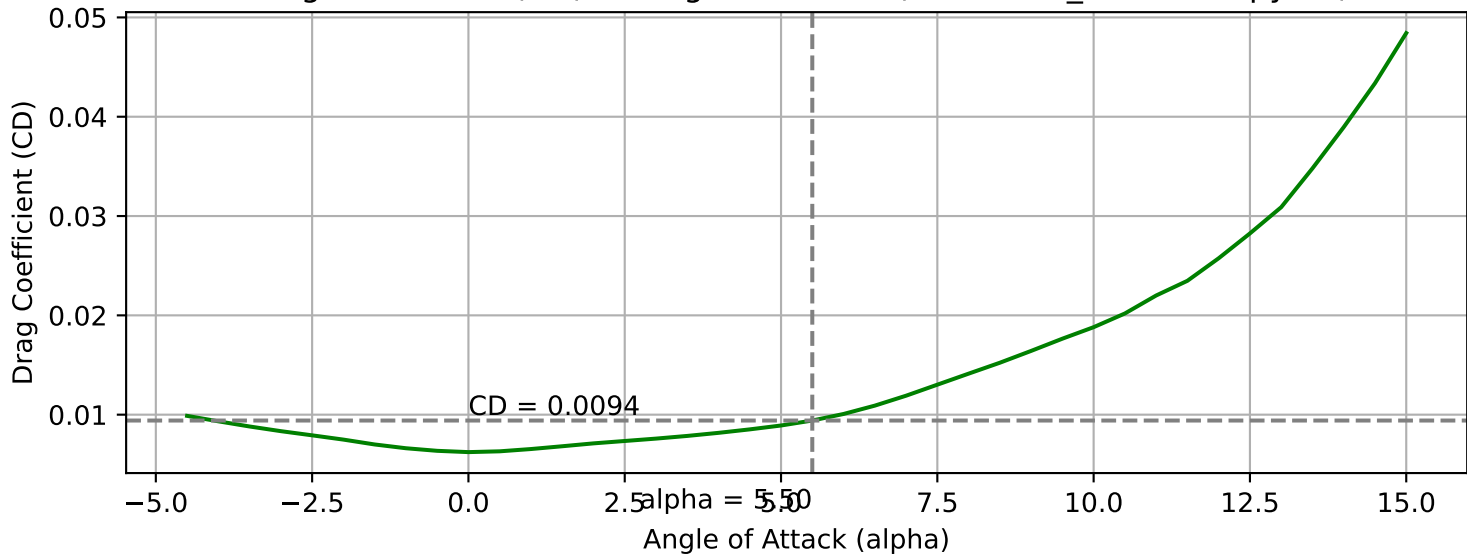
At  $\alpha = 5.50$ :  $C_L = 0.8504$ ,  $C_D = 0.0094$ , Max  $C_L/C_D = 90.3719$

# NACA 2412, $Re = 5.220e+05$

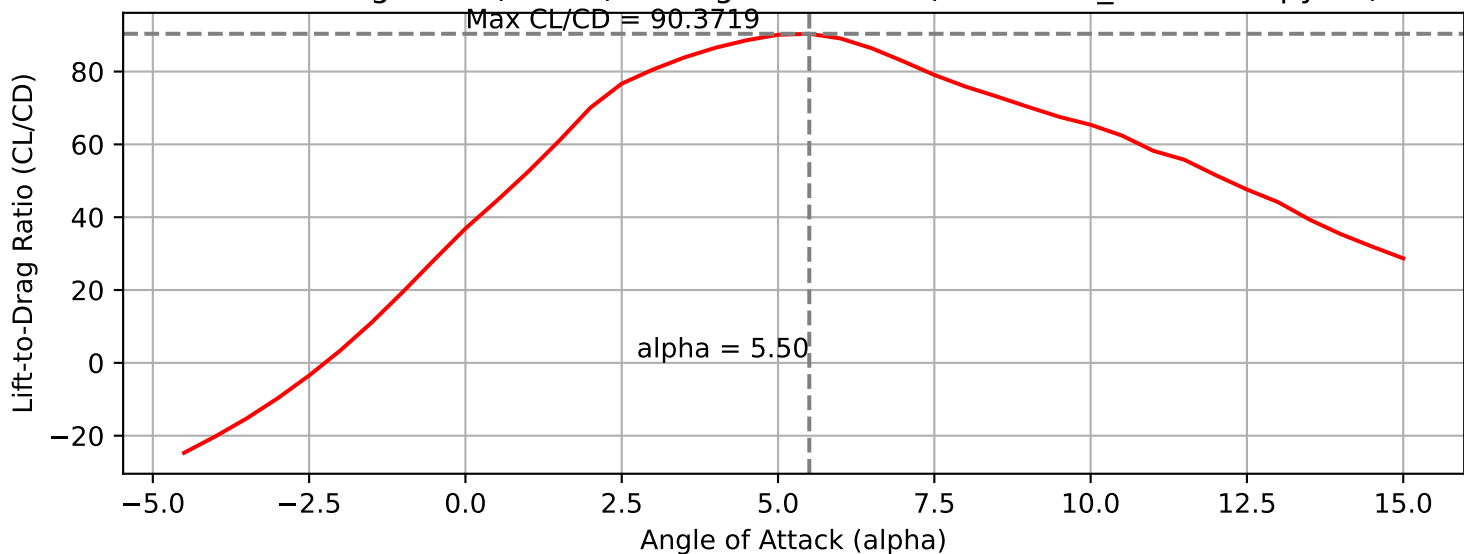
Lift Coefficient (CL) vs Angle of Attack (naca2412\_viscous copy.txt)



Drag Coefficient (CD) vs Angle of Attack (naca2412\_viscous copy.txt)



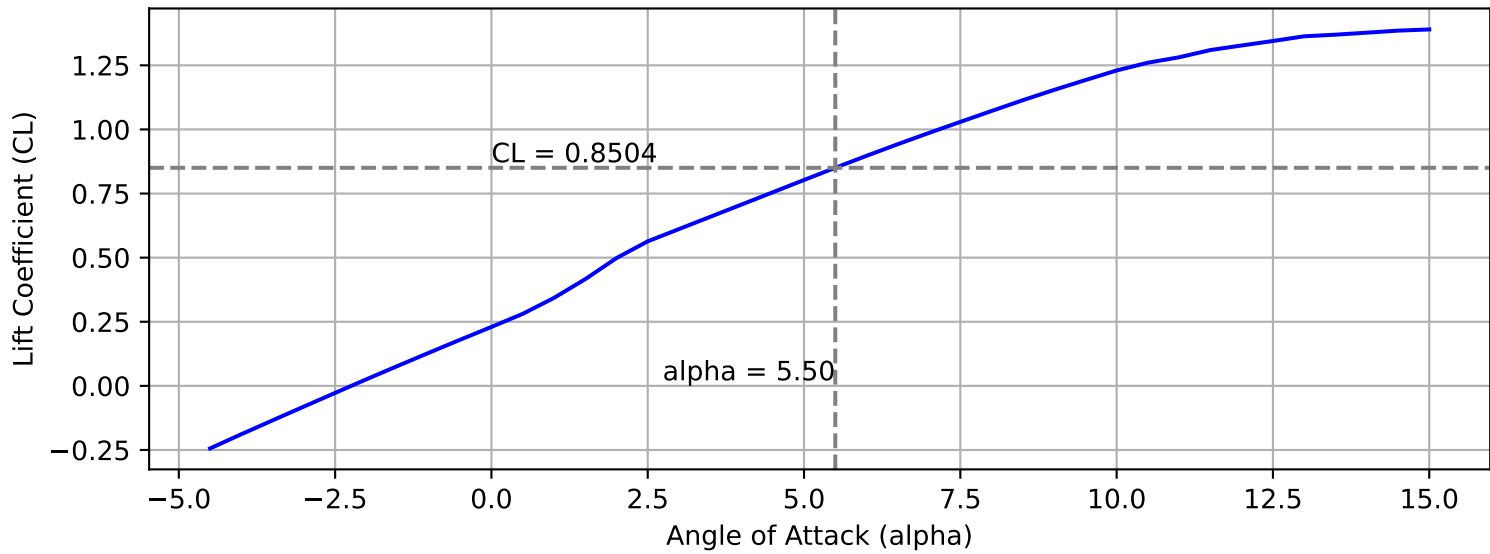
Lift-to-Drag Ratio (CL/CD) vs Angle of Attack (naca2412\_viscous copy.txt)



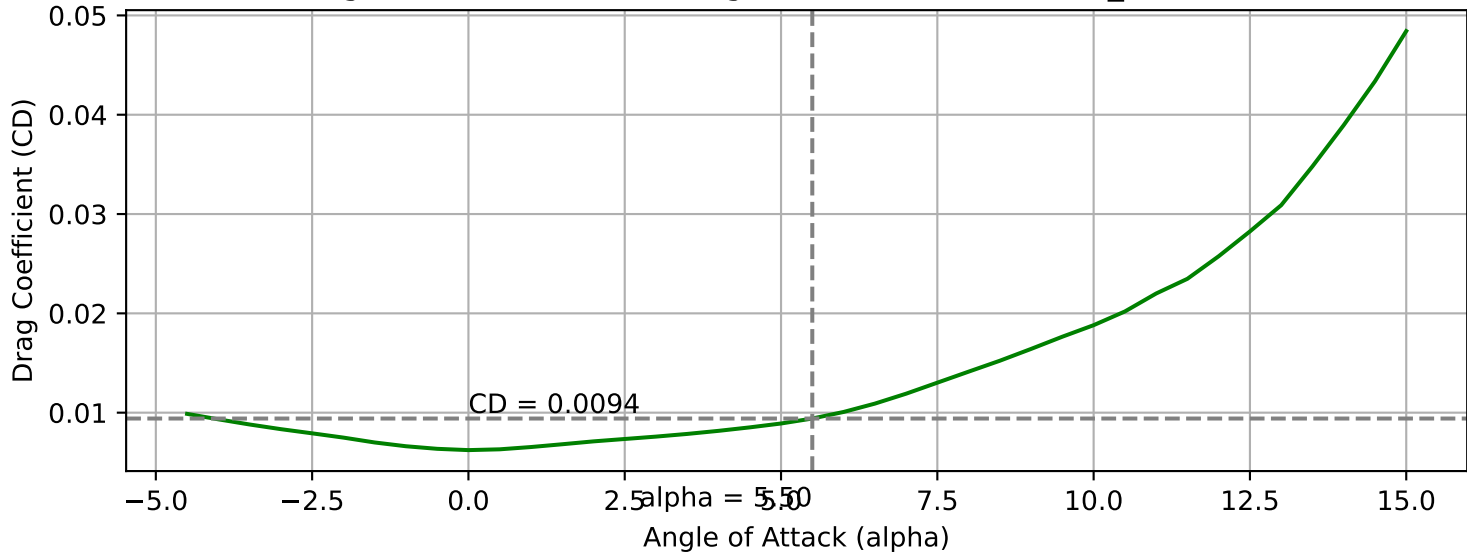
At alpha = 5.50: CL = 0.8504, CD = 0.0094, Max CL/CD = 90.3719

# NACA 2412, $Re = 5.220e+05$

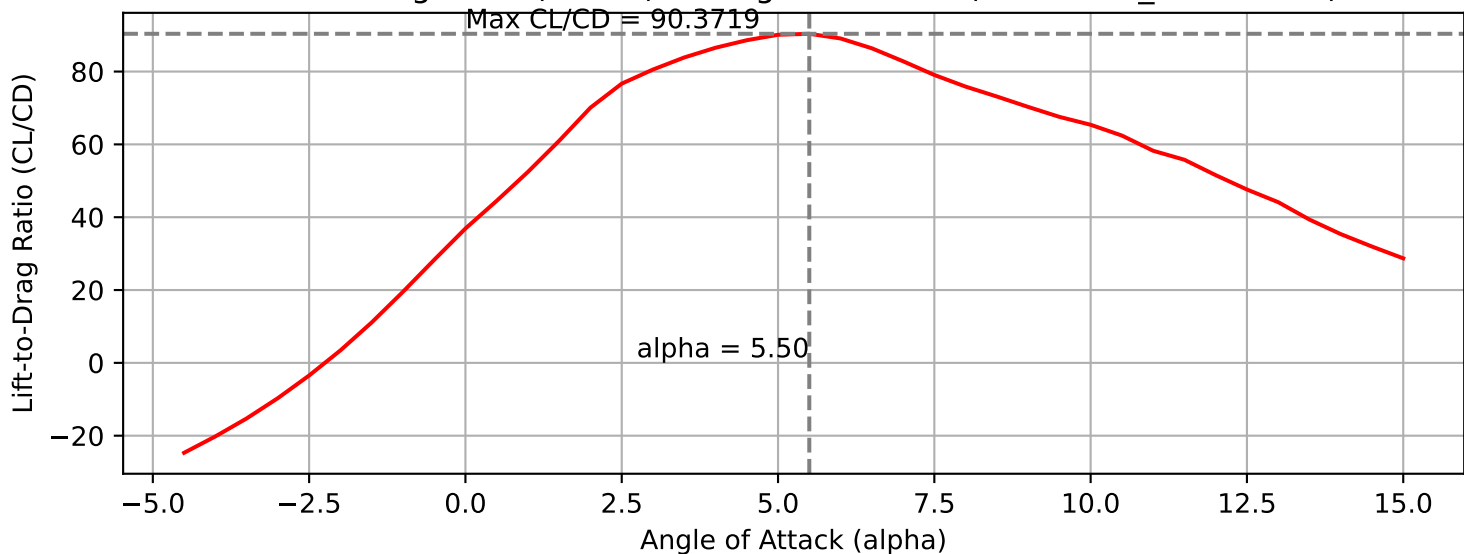
Lift Coefficient ( $C_L$ ) vs Angle of Attack (naca2412\_viscous.txt)



Drag Coefficient ( $C_D$ ) vs Angle of Attack (naca2412\_viscous.txt)



Lift-to-Drag Ratio ( $C_L/C_D$ ) vs Angle of Attack (naca2412\_viscous.txt)



At  $\alpha = 5.50$ :  $C_L = 0.8504$ ,  $C_D = 0.0094$ , Max  $C_L/C_D = 90.3719$