



TECHNISCHE  
UNIVERSITÄT  
WIEN  
Vienna University of Technology

VIENNA UNIVERSITY OF TECHNOLOGY

INSTITUTE OF APPLIED PHYSICS

SURFACE SCIENCE

---

# Bachelors Thesis

Title

---

*Author:*  
Martin Kronberger

*Supervisor:*

January 1, 2025



# Declaration of Authorship

**Martin Kronberger**

I hereby declare that I have written this thesis independently and that I have fully acknowledged all the sources and aids used. Furthermore, I confirm that I have marked as borrowed any parts of this work (including tables, maps, and figures) that are taken from other works or from the internet, either verbatim or in spirit, giving explicit reference to the source.

I further declare that large language models were used solely to assist with formatting, providing code snippets, and helping to resolve minor technical issues. These models did not contribute to the research content, analyses, interpretations, conclusions, or any intellectual substance of this thesis.

---

Martin Kronberger  
Vienna, 1 January 2025



## Acknowledgements



## Danksagung





## Abtract



## Kurzfassung



## Contents

<b>1</b>	<b>Foundational Principles</b>	<b>1</b>
1.1	Characteristic Length . . . . .	1
1.2	Dimension of the flow . . . . .	1
1.3	Turbulence and the Reynolds number . . . . .	2
1.4	Rarefaction and the Knudsen number . . . . .	2
1.5	Isentropic one-dimensional flow . . . . .	3
<b>2</b>	<b>Scope and objectives</b>	<b>5</b>
2.1	Geometry of the components . . . . .	5
2.2	Motivation and Goals . . . . .	7
<b>3</b>	<b>Analytical work</b>	<b>9</b>
3.1	Expected Flow Regimes . . . . .	9
3.2	One-dimensional isentropic variable area flow . . . . .	12
3.3	Further simplifications and impact of the leak . . . . .	14
3.4	Flow behaviors in micro channels . . . . .	16
3.5	Under-expanded nozzle plume at outlet . . . . .	17
<b>4</b>	<b>Discussion</b>	<b>18</b>
<b>5</b>	<b>Conclusion</b>	<b>19</b>
	<b>Symbols and Notation</b>	<b>21</b>
	<b>Formulary</b>	<b>22</b>
	<b>List of Figures</b>	<b>25</b>
	<b>List of Tables</b>	<b>25</b>
	<b>References</b>	<b>27</b>



---

## 1. Foundational Principles

### 1.1. Characteristic Length

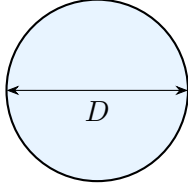
The characteristic length essentially serves the purpose of scaling physical systems. In dimensional analysis the goal is to find dimensionless quantities which describe the behavior of the system. These quantities are usually dependent on some characteristic scale, in our case the length scale, which describes the geometry of the model abstractly. For internal flows the characteristic length is defined as:

$$L_c = \frac{4A}{P_w} \quad \text{with} \quad P_w = \sum_{i=0}^{\infty} l_i$$

Where  $A$  is the cross-sectional area and  $P_w$  is the wetted perimeter, which is defined as the sum over the lengths of all surfaces in direct contact with the fluid. For gaseous fluids the whole perimeter of the cross-section must be considered, therefore the wetted perimeter reduces to the perimeter of the cross-section. The following section provides the characteristic length formulas for common duct shapes.

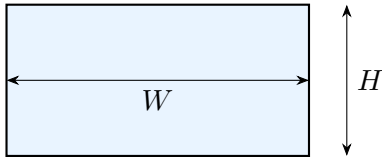
*Add some more context. Why do we need to parallel plates where can we use them?*

#### Circular duct or Nozzle



$$P_{w_o} = \pi D \quad \rightarrow \quad L_{c_o} = 2R$$

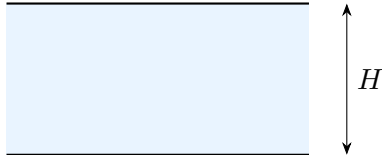
#### Rectangular duct



$$P_{w_{\square}} = 2H + W \quad \rightarrow \quad L_{c_{\square}} = \frac{2(HW)}{H + W}$$

#### Two parallel plates

These will be used in two different places.



$$\lim_{H \rightarrow \infty} L_{c_{\square}} = \lim_{H \rightarrow \infty} \frac{2(HW)}{H + W} = 2H$$

### 1.2. Dimension of the flow

The dimension describes the number of positional parameters needed to yield an exact solution for a given vector field  $V(\vec{x})$ , so equals essentially  $n = \dim(\vec{x})$ . The flow

through a constant area duct is usually described as a one-dimensional flow field only depending on the position  $x$  along the length of the duct. In the case of variable area ducts the flow will be three-dimensional and has to be calculated using all spatial coordinates. But assuming only a slight change in area along the length of the duct the flow can be approximated using a one-dimensional flow field with enough precision. This is called quasi one dimensional flow. [1]

### 1.3. Turbulence and the Reynolds number

### 1.4. Rarefaction and the Knudsen number

In gas dynamics a flow can be categorized by its particle interaction using the Knudsen number, which represents the ratio between the mean-free-path  $\lambda$  of the gas and some characteristic length  $L_c$ .

$$Kn = \frac{\lambda}{L_c}$$

The characteristic length is usually chosen to be the smallest linear length in the system. For example the throat diameter of a nozzle. [2]

#### Molecular regime ( $Kn \geq 10$ )

In this regime, the mean free path is much larger than the dimensions of boundaries. This leads to particle interactions themselves becoming negligible in comparison to the interaction of particles with the boundary. [3]

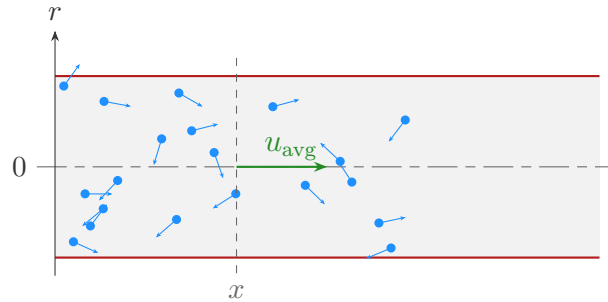


Figure 1: Mean velocity parallel to the flow at a point  $x$  inside a constant area duct in molecular flow.

#### Transition regime ( $0.1 \leq Kn \leq 10$ )

This regime is a middle ground between continuum and fully molecular flow. Neither the continuum assumptions of fluid dynamics nor the free molecular flow assumptions hold completely. The interactions between the gas molecules and the boundaries are significant, and the flow characteristics may vary widely.

#### Slip regime ( $0.001 \leq Kn \leq 0.1$ )

For increasing Knudsen numbers the mean free path becomes comparable to the characteristic length scale of the system. In this regime, the assumptions for continuum flow still hold, but there are deviations, especially near the boundaries. While continuum mechanics assumes no-slip conditions on the boundary, in this regime, slip on the boundary must be factored in.



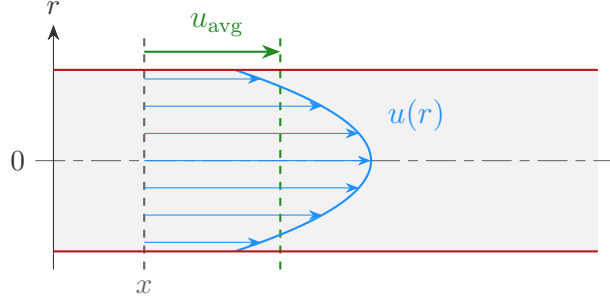


Figure 2: Velocity distribution at a point  $x$  inside a constant area duct with slip boundary conditions. [4]

### Continuum regime ( $Kn \leq 0.001$ )

In this regime, the interactions of particles in the medium are much more frequent than the interactions of particles with the boundaries of the duct. This makes it possible to describe the fluid itself as a continuous medium with the assumption of non-slip boundary conditions. The Navier-Stokes equations govern the calculations in this regime.

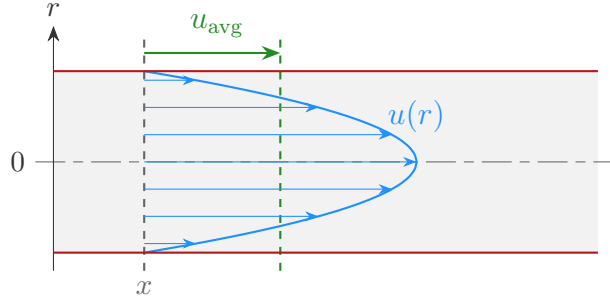


Figure 3: Velocity distribution at a point  $x$  inside a constant area duct with non-slip boundary conditions. (Continuum flow) [4]

### 1.5. Isentropic one-dimensional flow

Isentropic varying-area flow is one of the most idealized models to describe the behavior of gases flowing through a confined space. The following assumptions are made:

- steady, one-dimensional flow
- adiabatic:  $\delta q = 0, ds_e = 0$
- no shaft work:  $\delta w_s = 0$
- negligible change in potential energy:  $dz = 0$
- reversible:  $ds_i = 0$

Being reversible as well as adiabatic, the flow is therefore isentropic. The Mach number is defined as the ratio between the local velocity  $u$  and the local speed of sound  $a$ .

$$Ma = \frac{u}{a}$$

It is a very important metric when analyzing isentropic flow, since state variables are uniquely defined through the mach number at the corresponding location, as long as either the stagnation, or critical conditions of the flow are known.

**Low subsonic regime** ( $Ma < 0.3$ ) For low Mach numbers, compressibility effects of a gas can be neglected, and the gas can be treated as an incompressible fluid.

**Subsonic regime** ( $0.3 < Ma < 1.0$ ) Inside system of variable area ducts the gas flow generally stays subsonic. Once sonic speed is reached in a converging duct, the behavior reverses, and the velocity decreases, limiting the flow to subsonic or sonic speeds within converging ducts.

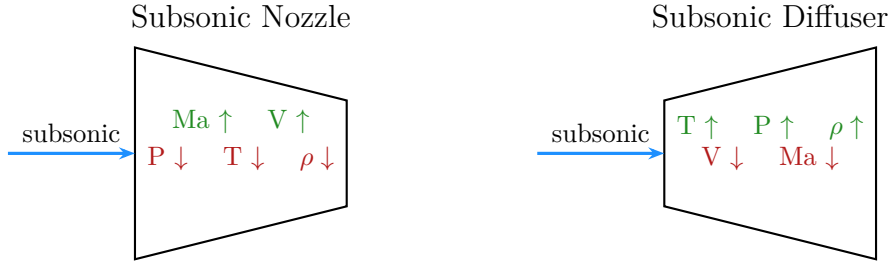


Figure 4: Change of flow properties in subsonic nozzles and diffusers

**Sonic regime** ( $Ma = 1$ ) Sonic flow occurs at the exit of a converging duct, if the pressure ratio between two reservoirs becomes smaller than the following critical ratio. Which is called choked flow and constitutes the maximum mass-flow for given stagnation conditions. This ratio is defined as:

$$\frac{P^*}{P_t} = \left( \frac{2}{\gamma + 1} \right)^{\gamma/(\gamma-1)}$$

Where  $P_t$  is the stagnation condition,  $P^*$  the critical back-pressure and  $\gamma$  the specific heat ratio. The ratio is derived from the isentropic flow relation (14) and can be expressed for any state variable.

**Supersonic regime** ( $Ma > 1$ ) If there are critical conditions at the end of a converging duct and a diverging duct follows. The flow continues to accelerate and reaches supersonic speeds. The location where the flow reaches critical condition is called throat and represents the minimal diameter of the duct.

In supersonic flows, state variables change rapidly causing phenomena like shock waves and expansion fans.

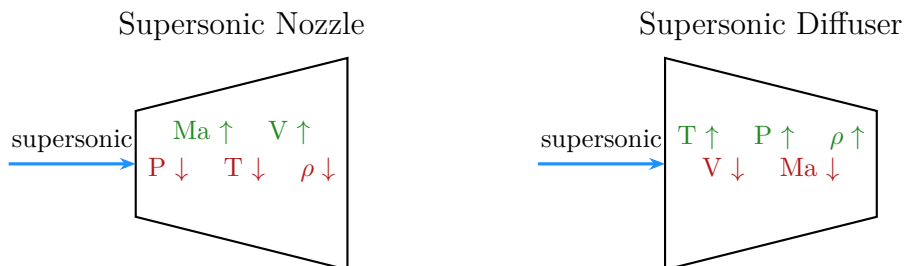


Figure 5: Change in flow properties in supersonic nozzles and diffusers

## 2. Scope and objectives

*Short introduction of the goal of the machine in general. What is it trying to do? Why do we even try to answer following questions? Maybe even try to focus on finding out if what is measured is even what we expect to measure. Simulations* To be able to establish reliable performance metrics for a given single atom catalyst

### 2.1. Geometry of the components

The geometry can be explained in three simple sections: gas from a reservoir flows over the inlet into the micro-reactor where it leaves through the outlet into a vacuum. This is a stark simplification, but for a great part of this thesis, this is how we will imagine our flow path. This is because the only thing we left out is any kind of leak in the system. Those leaks will be the most influential around the reactor, since this is the only part that is not held at a constant pressure by any external part of the system.

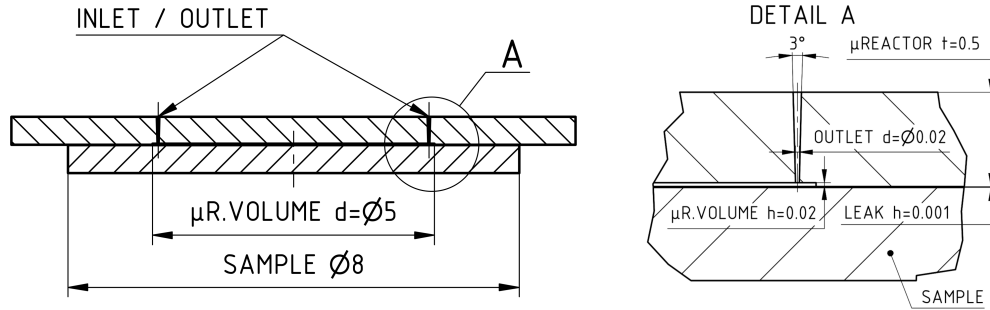


Figure 6: A descriptive caption for the figure.

#### Inlet Reservoir

It is kept at constant pressure  $P_0$  and constant temperature  $T_0$ . For simplicity it contains only one gas which is defined by its specific heat ratio  $\gamma$  and by its molar mass  $M_m$ . These are all parameters which are set in advance and will not change after being set.

#### Inlet Nozzle

The duct connecting the inlet reservoir with the reactor will be a slightly converging duct, due to production constraints. Therefore, it will act like a Nozzle, accelerating the gas until it expands into the reactor.

#### Reactor

The reactor resembles a very small but broad cylinder shape which is opened at the bottom. The sample will be pressed into the opening, which will lead to some leakage out of the system.

#### Outlet Nozzle

With the same geometry as the inlet, but the gas flowing in opposite directions, one would suspect the outlet to act as a subsonic nozzle, which could logically be the case, since without a converging section in front it would be impossible to reach sonic velocities and therefore will choke the flow and keep them at subsonic velocities. However,

it is actually possible for the flow to create a converging section by itself, which will force the flow to be sonic at the beginning of the outlet and will further accelerate into the supersonic regimes, creating a supersonic nozzle. Which of these two possibilities is most likely will be discussed at a later point.

### **Vacuum**

After leaving the outlet, the gas will first expand into a small cylindrical section after which it will expand freely into the vacuum. The exact pressure left in the vacuum will be very low, and small changes will not have great influence onto the flow itself. Therefore,

## 2.2. Motivation and Goals

The general goal of this thesis is to create a relatively simple analytical framework to be able to make predictions about the behavior of the flow through the system and approximate values at different positions in the flow to later be used as initial values for more complex numerical simulations. The following section will state specific questions we will then try to answer in the following sections.

### Type of Flow

The type of flow has major implications on which mathematical formulations and simulations are applicable, as well as the way the gas particles interact with each other and the walls of the assembly. The main focus here is the Knudsen number and the idealized flow regimes connected to it. With the main goal being to assess the most likely flow regime governing the inside of the assembly and therefore determine the equations applicable to calculate the state variables at different points in the system and the throughput of the system as a whole.

In preparation for numerical simulations it is also important to find a way to calculate Knudsen numbers and other flow parameters using given datasets of state variables without having to rely on flow regime specific methods. This will help to analyze transient regimes, encountered when the gas expands into the vacuum, using one generally applicable method.

### Impact of the leak

As described in the leading section there will be some leakage expected at the boundary between the reactor casing and the sample holder. This leak will inevitably lead to some mass-flow and therefore some pressure drop  $\Delta P_L$  inside the reactor. This can lead to major changes to the steady state of the system, therefore influencing the velocity distribution at the outlet. In summary the goal is finding the pressure drop  $\Delta P_L$  caused by the leak and the effective mass flow  $\dot{m}_L$  through it and use them to predict probable behaviors.

### Behavior of the gas around the sample

Knowing more about the state the gas is in close to the sample is very helpful. These can be used to estimate values for diffusion rates to and from the surface, boundary layer thickness, and mean velocities and momentum of particles reaching the reactive surface. Important metrics are the velocity, pressure and temperature of the gas and the type of flow.

### Velocity distribution at the outlet

After the gas leaves the outlet nozzle, it expands into a vacuum chamber where the gas atoms are ionized by an electron beam and picked up by the mass-spec, to measure the ratio of different products. This won't be the case for all atoms, since for an atom to be ionized it has to cross the electron beam, which is localized in space. The remaining gas has to be pumped out and doesn't contribute to the ratio measured by the mass-spec. Therefore, it is important to approximate, how much of the gas leaving the outlet actually is able to reach the region of influence of the electron beam and will contribute to the measurement of the mass-spec. To answer this the velocity distribution of the expansion after the gas is fully rarefied is needed. This distribution can then essentially be treated as a source like surface, with no interaction between gas particles, and can

therefore be directly correlated to the amount of atoms reaching the sphere of influence of the electron beam.

Knowing what determines the distribution of the outlet can also help identify changes to be made to the geometry or the reservoir conditions to increase the amount of atoms reaching the mass-spec.

### 3. Analytical work

*We will be using nitrogen as our gas for the whole section, therefore  $\gamma = 1.470$  and  $M = 28.013\text{g/mol}$*

#### 3.1. Expected Flow Regimes

##### Continuum Regime

The whole Theory relies heavily on the continuum model. For every location of the flow, but the vacuum, it will suffice to approximate the state variables at certain locations using a simple continuum model and use these state variables to calculate the Knudsen number.

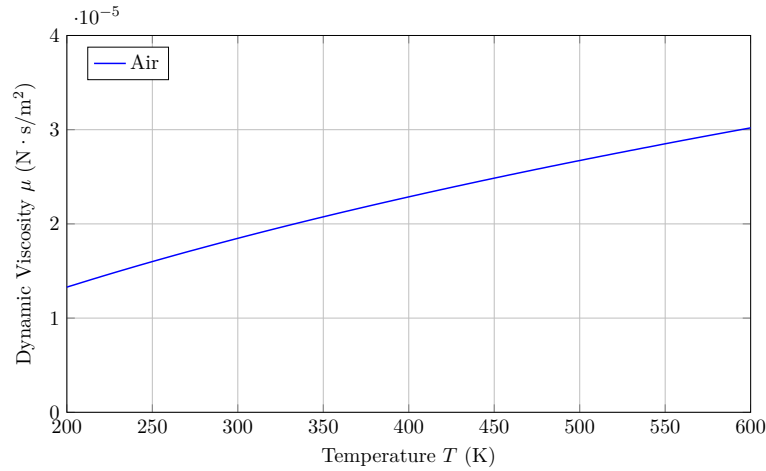
A sensible question to ask at this point is: Where in the flow it is the most probable encounter the highest Knudsen number. Because this will reduce the locations to check for high Knudsen numbers and therefore will simplify the process. To find such most probable Location, given the Definition of the Knudsen number:

$$Kn(p, T) = \frac{\lambda}{L_c} = \frac{\mu(T)R}{pL_c} \sqrt{\frac{\pi m T}{2k_B}}$$

Where  $\lambda$  is the mean free path,  $L_c$  is the characteristic length,  $k_B$  is the Boltzmann constant,  $R$  is the specific gas constant,  $T$  is the temperature of the fluid,  $p$  is the pressure of the fluid,  $m$  is the molecular mass and  $\mu$  is the dynamic viscosity.

The dynamic viscosity can be calculated using Sutherland's formula.

$$\mu(T) = \mu_0 \left( \frac{T}{T_0} \right)^{3/2} \frac{T_0 + S_\mu}{T + S_\mu}$$



$L_c$  can be assumed constant since the height of the reactor and the smallest diameter of the ducts match. And by assuming  $\mu$  to be close to constant the Knudsen number becomes a simple proportionality relation.

$$Kn(p, T) \approx \frac{\mu R}{L_c} \sqrt{\frac{\pi m}{2k_B}} \cdot \frac{\sqrt{T}}{p} = \alpha \cdot \frac{\sqrt{T}}{p} \rightarrow Kn \propto \frac{\sqrt{T}}{p}$$

This makes it obvious that areas with low pressure will lead to higher Knudsen numbers and will come closest to the limit of  $Kn = 0.1$  where continuum regime formulations stop to yield sensible solutions. Therefore, calculating the Knudsen number where the

gas leaves the outlet nozzle will be useful to identify the flow regime that governs the gas flow inside the whole assembly.

*This argument makes more sense than calculating some ratio which can be tested, since it is the same as just checking Knudsen number. This is something we can always check during the calculations, since we will calculate  $p$  and  $T$  anyway. But we can still improve this argumentation and maybe state that checking the Knudsen number at different sections makes sense, but only while calculating, not in advance.*

**Knudsen Number in low pressure Zones** As the gas is leaving the outlet geometry and expands into the vacuum the characteristic length loses its significance. This is because the walls of the vacuum chamber are very far away in comparison to the length-scales of the flow geometry, while the gas expands it will lose pressure to conform to the vacuum and in that process will transition into free molecular flow. This leads to formulations using the mach number also losing its significance. Therefore, it makes sense to identify a much more elegant way of calculating the local Knudsen number  $Kn_L$  which will be much more applicable in this situation.

$$Kn_L = \frac{\lambda}{\phi} \left| \frac{d\phi}{dx} \right|$$

Where  $\lambda$  is the mean free path,  $\phi$  is some state variable of the flow. This way the Knudsen number can be calculated throughout the expansion of the gas and can be used to find the contour lines where the transition between continuum and molecular flow will happen. *Add reference!*

### Laminar Flow

The Reynolds number itself does not bare any real significance for the applicability of the formulations used. Nonetheless, deviations in Reynolds number can in reality shape great parts of how the flow behaves and therefore effects important state variables. Reynolds numbers per unit length for isentropic expansion range  $10^{-2} < Re/l < 1$ , whereas the characteristic length is constant at  $L_c = 20 \cdot 10^{-6}$  and will therefore dominate the equation forcing low Reynolds numbers. [5]

Therefore the flow will stay laminar as long as it is contained inside the assembly and therefore mixing will be most likely diffusion mediated. [6]

### Dimension of the Flow

**Inside - pseudo 1D** In this section of the flow a further simplification of the two-dimensional flow through a duct will be used called "quasi one-dimensional" flow. This is possible by reducing the velocity distribution present at any point in a duct to its mean velocity. Therefore, reducing the velocity from a distribution  $V(r)$  at every point in the duct to a scalar value  $V$  at every point of the duct. This is a general simplification which can be made when using continuum flow analysis inside of ducts and is only bound by the applicability of continuum flow. This makes sense, because approximations of state variables and mass flow is the main goal here. Taking mean values still yield sensible solutions and won't leave us with less information.

**Outside - radial symmetric 3D** Any further simplification from the radially symmetric expansion into vacuum described in section would not make sense since it would eliminate important information on how the flow behaves when leaving the outlet. Like the amount of gas which will reach the mass spec and the velocity distribution. [1]



**Steady Flow**

Steadiness of the flow is essentially given by the fact that the temperature  $T_0$  and the pressure  $p_0$  of the reservoir will be held constant during measurements. Thus, the flow will be driven by the pressure differential between the vacuum and the reservoir alone. So there is no reason for the flow to establish any dynamic behaviors after reaching some equilibrium state, after gas is first released into the system. This is true for regions inside as well as outside the assembly.

### 3.2. One-dimensional isentropic variable area flow

By assuming the flow through the assembly will be fully isentropic and pseudo one-dimensional it is possible to calculate the state variables at every point knowing the stagnation conditions and the ratio between the cross-sectional area at the point of interest  $A_i$  and the at throat of the assembly  $A^*$ .

It must be noted that this is a very radical approximation since for the flow to be considered pseudo one-dimensional the problem must be reduced to consist purely of variable area ducts. This clearly overlooks the fact that when entering and leaving the reactor the gas has to perform a right angle turn to follow the flow path. Another constraint on the duct geometry to achieve reasonable solutions assuming pseudo one-dimensional flow is that the duct must change its cross-sectional area gradually. [1] This won't be the case inside the reactor, since there is no way of slicing the reactor chamber to achieve a gradual change in cross-section, especially around the inlet and outlet.

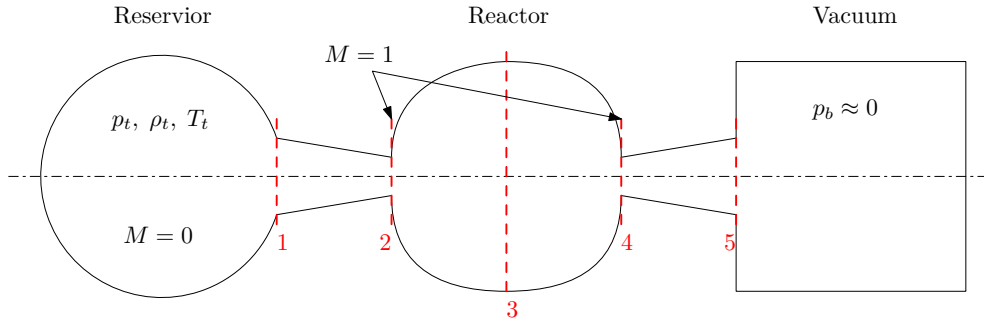


Figure 7: A descriptive caption for the figure.

This geometry can now be characterized as a double throat, and therefore isentropic flow can result in one of two fundamental solutions. One predicting subsonic flow and one predicting supersonic flow after the first throat (2). [7] [8]

#### Calculations

The first step is to define the critical locations, where the flow will be choked. Since the outlet is expanding into vacuum, resulting in a pressure ratio tending towards zero, therefore the flow therefore must be choked and can be recognized as a critical point. Maximum mass-flow occurs if the flow is choked. Therefore, to keep up with the mass-flow of the outlet, the inlet must also be choked:

$$A_{2,4}, p_{2,4}, \rho_{2,4}, T_{2,4} \xrightarrow{M=1} A^*, p^*, \rho^*, T^*$$

The second reference location corresponding to the stagnation or total conditions is at the entry of the inlet nozzle (1), which can be defined afterward to get quantitative solutions.

$$A_1, p_1, \rho_1, T_1 \xrightarrow{M=0} A_t, p_t, \rho_t, T_t$$

Next step is to calculate the cross-sectional areas for every location:

$$A_i = \pi \left( \frac{D_i}{2} \right)^2 \quad \text{for } i = \{1, 2, 4, 5\} \quad \text{and} \quad A_3 = H_{\text{reactor}} \cdot D_{\text{reactor}}$$

Followed by solving the equation for the ratio of cross-sectional area for  $M$  numerically which, should yield one subsonic and one supersonic solution.

$$\frac{A}{A^*} = \frac{1}{M} \left[ \frac{2}{\gamma + 1} \left( 1 + \frac{\gamma - 1}{2} M^2 \right) \right]^{\frac{\gamma + 1}{2(\gamma - 1)}} \quad (10)$$

Afterward the ratios of state variables can be determined, which after defining the total conditions can be used to calculate the local variables for every point. [9]

$$\frac{T}{T_t} = \left( 1 + \frac{\gamma - 1}{2} M^2 \right)^{-1} \quad (14)$$

$$\frac{p}{p_t} = \left( 1 + \frac{\gamma - 1}{2} M^2 \right)^{-\frac{\gamma}{\gamma - 1}} \quad (15)$$

$$\frac{\rho}{\rho_t} = \left( 1 + \frac{\gamma - 1}{2} M^2 \right)^{-\frac{1}{\gamma - 1}} \quad (16)$$

Where  $p$ ,  $\rho$ ,  $T$  are the local gas conditions,  $p_t$ ,  $\rho_t$ ,  $T_t$  the total gas conditions,  $\gamma$  the specific heat ratio and  $M$  the local mach number. This leads to the following solutions:

$i$	$\frac{A_i}{A^*}$	M	$\frac{p_i}{p_t}$	$\frac{\rho_i}{\rho_t}$	$\frac{T_i}{T_t}$
1	4	0	1	1	1
2	1	1	0.52	0.64	0.81
3	318.31	$\sim 0$	$\sim 1$	$\sim 1$	$\sim 1$
		10.55	$3.28 \cdot 10^{-5}$	$8.90 \cdot 10^{-4}$	$3.68 \cdot 10^{-2}$
4	1	1	0.52	0.64	0.81
5	4	0.15	0.983	0.988	0.994
		3.06	0.02629	0.08415	0.31245

Table 1: Isentropic flow properties for different conditions.

Mass flow must be conserved along the flow and can be calculated using following equation, which derives from the general equation for mass-flow, the isentropic relations and the ideal gas law. [10]

$$\dot{m} = A \cdot P_t \cdot \sqrt{\frac{\gamma}{RT_t}} \cdot M \cdot \left( 1 + \frac{\gamma - 1}{2} M^2 \right)^{-\frac{\gamma + 1}{2(\gamma - 1)}}$$

Where  $A$  is the local cross-sectional area,  $P_t$  the total pressure,  $T_t$  the total temperature,  $\gamma$  the specific heat ratio,  $R$  the specific gas constant and  $M$  the mach number. [11]

### Interpretation

*Go more into the solutions. Especially into the non physicality of the high mach numbers inside. Also mention non isentropic processes. Calculate the mass flow for all the points, to check validity.*

### 3.3. Further simplifications and impact of the leak

In section 3.2 the flow is assumed to be fully isentropic and one-dimensional. This led to the simplification of the reactor being a variable area duct, which leads to a change in state-variables along the path of the gas flowing from the inlet to the outlet. Since this change is based on the variable area duct formulation, it won't represent an accurate solution for the given geometry. The leak is located at the perimeter of the reactor, which is not an actual location in the current formulation. To be able to include the influence of the leak analytically, without using numerical tools, further simplification is needed.

Thus, this section will explore a formulation of the system, where both the inlet reservoir and the reactor are assumed to be reservoirs, with stagnation conditions associated with them. In turn disconnecting the two reservoirs and resulting in two independent problems, describing gas from a reservoir flowing into a discharge region with a given back pressure. Representing a change in stagnation conditions which can be attributed to non-isentropic processes happening when the gas enters or leaves the reactor.

Now proper assumptions have to be taken to connect both reservoirs again, first without inclusion of the leak. Afterward two possible formulations including the leak will be presented.

#### Formulation without the leak

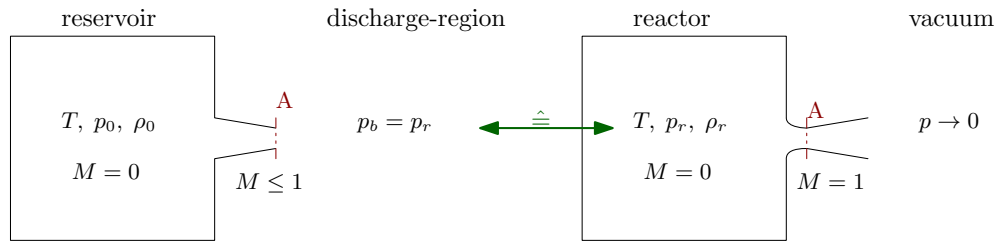


Figure 8: A descriptive caption for the figure.

#### Assumptions

Since this still constitutes a closed system, mass flow must be conserved, thus:

$$\dot{m}_0 = \dot{m}_r = \text{const.}$$

The reactor still discharges into vacuum, this therefore must lead to choked flow:

$$M_r = 1$$

From the geometry it is given that:

$$A_0 = A_r = A$$

*Explain why the temperature is held constant in more detail.*

$$T_r = T_0 = T$$

The mass flow rates from the reactor into vacuum, can be calculated using the isentropic mass flow equation:

Reservoir  $\rightarrow$  Reactor:

$$\dot{m}_0 = A p_0 \sqrt{\frac{\gamma}{RT}} M_0 \left( 1 + \frac{\gamma-1}{2} M_0^2 \right)^{-\frac{\gamma+1}{2(\gamma-1)}}$$

Reactor  $\rightarrow$  Vacuum:

$$\dot{m}_r = A p_r \sqrt{\frac{\gamma}{RT}} \left( 1 + \frac{\gamma-1}{2} \right)^{-\frac{\gamma+1}{2(\gamma-1)}}$$

Thus

$$p_r \left( \frac{2}{\gamma+1} \right)^{\frac{\gamma+1}{2(\gamma-1)}} = p_0 M_0 \left( 1 + \frac{\gamma-1}{2} M_0^2 \right)^{-\frac{\gamma+1}{2(\gamma-1)}}.$$

Using the isentropic relation

$$p_0 = p_r \left( 1 + \frac{\gamma-1}{2} M_0^2 \right)^{\frac{\gamma}{\gamma-1}}$$

Then

$$0 = M_0 \left( 1 + \frac{\gamma-1}{2} M_0^2 \right)^{\frac{1}{2}} - \left( \frac{2}{\gamma+1} \right)^{\frac{\gamma+1}{2(\gamma-1)}}$$

### 3.4. Flow behaviors in micro channels

Going from macro scale channels to microscales has some major implications for the behavior of the gas. The primary factor for these differences is slipping at the boundary of the surfaces. This is due to the fact that at small characteristic length scales the Knudsen number ( $Kn$ ), whose value describes the interaction of the molecules in the gas and its boundaries, becomes relatively high ( $Kn > 0.001$ ). Which usually puts the gas flow in the category of compressible flow with slip at the boundaries.

Most of these behaviors have to be studied using complex simulations or experimental results and are sometimes not fully explained. Therefore, this section is not intended to give concrete definitions or formulations, but to provide as many relevant references regarding behaviors in microfluidics as possible. It will be mainly based on the review study conducted by Amit Agrawal. [12]

#### Phenomenon of Slip

Slip, in comparison to non-slip, refers to the fact that the tangential velocity close to a surface is non-zero. Determining this velocity at the boundary is the main focus when slip is to be included in future calculations.

**Slip models** Maxwell suggested that on a control surface  $s$ , at distance half the mean free path away from the surface, one half of the molecules come in from one mean free path away with the tangential velocity  $u_\lambda$ , the other half is reflected from the surface. Assuming a fraction  $\sigma$  of the molecules are reflected diffusively (average velocity corresponds to velocity at the wall  $u_w$ ) at the walls, with the remainder  $(1 - \sigma)$  being reflected specularly (no change in their impinging velocity  $u_\lambda$ ). When expanding  $u_\lambda$  in a second order Taylor series this yields:

$$u_g - u_w = \left[ \frac{2 - \sigma}{\sigma} Kn \left( \frac{\partial u}{\partial n} \right)_s + \frac{Kn^2}{2} \left( \frac{\partial^2 u}{\partial n^2} \right)_s \right]$$

Where  $u$  stands for the streamwise velocities, where the subscripts  $g$ ,  $w$  and  $s$  refer to gas, wall and control surface, with  $n$  being the normal to the control surface. And most importantly  $\sigma$  is the tangential momentum accommodation coefficient, or short TMAC.

The TMAC can be expressed in terms of tangential momentum of the incident molecules  $\tau_i$  and the tangential momentum of the reflected ones  $\tau_r$ :

$$\sigma = \frac{\tau_i - \tau_r}{\tau_i}$$

Determining the TMAC is

#### Surface Roughness

#### Numerical simulations

#### Sudden expansion or contraction

### 3.5. Under-expanded nozzle plume at outlet

After the gas reaches sonic speeds at the throat of the outlet nozzle, it expands in the diverging outlet nozzle. This leads to a steady pressure drop as the gas reaches the exit plane of the nozzle. There are three distinctive expansion patterns characterized by the difference in pressure between the exit of the nozzle and the back pressure of the chamber.

If these values match, this is called fully expanded, which leads to a straight column of gas leaving the nozzle and no shock waves being created. If the pressure of the gas leaving the nozzle is lower than the back-pressure. The gas inside gives way to the back-pressure, leading to oblique shockwaves forming at the exit, compressing the column of gas, this is called under-expanded. If the back-pressure is now lower than the pressure of the gas, in this over-expanded state the gas leaving the nozzle further expands over the edges of the exit, creating what is called expansion fans.

These fans are often called prandtl-meyer fans and occur when super-sonic flow has to turn around a sharp edge. *brandl-meyer angle and maximum angle. More information about connection with characteristic lines, also explanation on the process of rarification and transistion regime.*

#### Method of characteristics

**Use for nozzle design** The method of characteristics is a mathematical technique used to design supersonic nozzles so that gas flows expand smoothly from sonic to supersonic speeds without generating internal shocks. It works by tracing characteristic lines—paths along which flow properties remain constant—through the nozzle region. By aligning the walls with these lines it ensures that each incremental flow turn occurs through a series of controlled expansion waves, rather than abrupt angle changes that could cause shocks. If fully expanded this leads to a straight column of gas leaving the nozzle, where all of its energy is converted into kinetic energy without significant losses due to shocks.

#### Navier-Stokes Equations

*Fundamental equations for solving continuum flow problems. No information about boundary layer thickness and boundary conditions. These need simulations like DSMC to define boundary layer*

#### Direct simulation Monte Carlo (DSMC)

*Fundamentals of montecarlo simulations in general. How does it solve continuum regime even tho it uses molecular simulations. Use for boundary layer approximations.* [2]

[13] [14]

## 4. Discussion

*Further work to be done:*

- *Better formulation of the leakage.*
-



---

## 5. Conclusion



---

## Symbols and Notation

### Primary Symbols and Definitions

$a$	speed of sound	$u$	internal energy per unit mass
$A$	cross-sectional area of a duct	$v$	specific volume, $\frac{1}{\rho}$
$\xi$	pressure ratio across a shock wave, $\frac{p_2}{p_1}$	$u$	velocity components parallel to flow
$\rho$	mass density	$v$	velocity components perpendicular to flow
$c_p$	specific heat at constant pressure	$V$	speed of flow
$c_v$	specific heat at constant volume	$V_\infty$	maximum speed at absolute zero
$h$	enthalpy per unit mass, $u + pv$	$\beta$	$\sqrt{M^2 - 1}$
$L_c$	characteristic reference length	$\gamma$	$\frac{c_p}{c_v}$ (ratio of specific heats)
$M$	Mach number, $\frac{V}{a}$	$\theta$	shock-wave angle from upstream direction
$p$	pressure	$\mu$	Mach angle, $\sin^{-1} \frac{1}{M}$
$q$	dynamic pressure, $\frac{\rho V^2}{2}$	$\mu$	absolute viscosity
$R$	gas constant	$\nu$	Prandtl-Meyer angle
$s$	entropy per unit mass		
$T$	absolute temperature		

### Subscripts

$\infty$	free-stream conditions
$t$	total conditions (isentropic rest conditions)
$*$	critical conditions (local speed equals local sound speed)
$r$	reference (or datum) values
perf	thermally and calorically perfect gas
therm perf	thermally perfect, calorically imperfect gas

### Notations

[perf]	thermally and calorically perfect gas.
[therm perf]	thermally perfect but calorically imperfect gas.
[isen]	isentropic flow process (not valid for shock waves).
[adiab]	adiabatic process (no heat transfer, may or may not be isentropic).

**Formulary****Thermodynamics****Thermal Equations of State**

Is in general an equation of the form

$$p = p(v, T) \quad (1)$$

Commonly used equations of state for thermally perfect gases

$$p = \frac{RT}{\rho} \quad [\text{therm perf}] \quad (2)$$

$$\frac{1}{\rho} dp + V dV = 0 \quad [\text{therm perf}] \quad (3)$$

**Control Volume Analysis****Continuity Equation:**

$$\frac{d}{dt} \int_V \rho dV + \int_S \rho V \cdot dA = 0 \quad (4)$$

**Momentum Equation:**

$$\frac{d}{dt} \int_V \rho V dV + \int_S \rho V (V \cdot dA) = - \int_S p dA + \int_V \rho g dV \quad (5)$$

**Energy Equation:**

$$\frac{d}{dt} \int_V \rho e dV + \int_S \rho h V \cdot dA = \dot{Q} - \int_S p V \cdot dA \quad (6)$$

**Continuous one-dimensional****Speed of Sound:**

$$a = \sqrt{\left(\frac{\partial p}{\partial \rho}\right)_s} = \sqrt{\gamma \left(\frac{\partial p}{\partial \rho}\right)_T} \quad (7)$$

$$a = \sqrt{\gamma \frac{p}{\rho}} = \sqrt{\gamma R T} \quad [\text{therm perf}] \quad (8)$$

**Mach Number:**

$$M = \frac{V}{a} \quad (9)$$

**Area-ratio in terms of mach number:**

$$\frac{A}{A^*} = \frac{1}{M} \left[ \frac{2}{\gamma + 1} \left( 1 + \frac{\gamma - 1}{2} M^2 \right) \right]^{\frac{\gamma + 1}{2(\gamma - 1)}} \quad [\text{isen, perf}] \quad (10)$$

---

**Isentropic Relations:**

$$\frac{p}{\rho^\gamma} = \text{constant} = \frac{p_t}{\rho_t^\gamma} \quad [\text{isen, perf}] \quad (11)$$

**From this:**

$$\frac{p}{p_t} = \left(\frac{\rho}{\rho_t}\right)^\gamma = \left(\frac{T}{T_t}\right)^{\frac{\gamma}{\gamma-1}} = \left(\frac{a}{a_t}\right)^{\frac{2\gamma}{\gamma-1}} \quad [\text{isen, perf}] \quad (12)$$

**Bernoulli's Equation for Compressible Flow**

$$\frac{\gamma}{\gamma-1} \left(\frac{p_t}{\rho_t}\right)^{\frac{\gamma-1}{\gamma}} \left(\frac{p}{p_t}\right)^{\frac{1}{\gamma}} + \frac{V^2}{2} = \frac{\gamma}{\gamma-1} \frac{p_t}{\rho_t} \quad [\text{isen, perf}] \quad (13)$$

**Stagnation Relations**

$$\frac{T}{T_t} = \left(1 + \frac{\gamma-1}{2} M^2\right)^{-1} \quad [\text{adiab, perf}] \quad (14)$$

$$\frac{p}{p_t} = \left(1 + \frac{\gamma-1}{2} M^2\right)^{-\frac{\gamma}{\gamma-1}} \quad [\text{isen, perf}] \quad (15)$$

$$\frac{\rho}{\rho_t} = \left(1 + \frac{\gamma-1}{2} M^2\right)^{-\frac{1}{\gamma-1}} \quad [\text{isen, perf}] \quad (16)$$

$$\frac{a}{a_t} = \left(1 + \frac{\gamma-1}{2} M^2\right)^{-\frac{1}{2}} \quad [\text{adiab, perf}] \quad (17)$$



**List of Figures**

1	Mean velocity parallel to the flow at a point $x$ inside a constant area duct in molecular flow. . . . .	2
2	Velocity distribution at a point $x$ inside a constant area duct with slip boundary conditions. [4] . . . . .	3
3	Velocity distribution at a point $x$ inside a constant area duct with non-slip boundary conditions. (Continuum flow) [4] . . . . .	3
4	Change of flow properties in subsonic nozzles and diffusers . . . . .	4
5	Change in flow properties in supersonic nozzles and diffusers . . . . .	4
6	A descriptive caption for the figure. . . . .	5
7	A descriptive caption for the figure. . . . .	12
8	A descriptive caption for the figure. . . . .	14

**List of Tables**

1	Isentropic flow properties for different conditions. . . . .	13
---	--	----





## References

1. ANDERSON, John D. *Modern Compressible Flow: With Historical Perspective*. 4th. New York: McGraw-Hill Education, 2021. ISBN 9781260570556.
2. PUTIGNANO, Massimiliano. *Supersonic Gas-Jet Based Beam Profile Monitor*. Liverpool, UK, 2012. PhilosophiæDoctor (PhD). University of Liverpool.
3. RAPP, Bastian E. *Microfluidics: Modeling, Mechanics and Mathematics*. Oxford, UK: Elsevier, 2017. ISBN 978-1-4557-3141-1. Available from DOI: 10.1016/C2012-0-02230-2.
4. ÇENGEL, Yunus A.; CIMBALA, John M. *Fluid Mechanics: Fundamentals and Applications*. 4th. New York, NY: McGraw Hill, LLC, 2017. Mechanical Engineering.
5. AMES RESEARCH STAFF. *Equations, Tables, and Charts for Compressible Flow*. Moffett Field, CA, 1953. Tech. rep., Report 1135. National Advisory Committee for Aeronautics. Available also from: <https://ntrs.nasa.gov/api/citations/19930091059/downloads/19930091059.pdf>.
6. COMSOL. *Microfluidics Module User's Guide*. Stockholm, Sweden: COMSOL AB, 2022. Version: COMSOL 6.1, Part number: CM021901. Available at: [www.comsol.com](http://www.comsol.com).
7. SALAS, M.D.; ABARBANEL, S.; GOTTLIEB, D. Multiple steady states for characteristic initial value problems. *Applied Numerical Mathematics*. 1986, vol. 2, no. 3, pp. 193–210. ISSN 0168-9274. Available from DOI: [https://doi.org/10.1016/0168-9274\(86\)90028-0](https://doi.org/10.1016/0168-9274(86)90028-0). Special Issue in Honor of Milt Rose's Sixtieth Birthday.
8. *Fundamentals of Gas Dynamics* [online]. Princeton University Press, 1958 [visited on 2025-01-21]. ISBN 9780691626499. Available from: <http://www.jstor.org/stable/j.ctt183pg9x>.
9. HALL, Nancy. *Isentropic Flow Equations* [online]. [N.d.]. [visited on 2024-07-18]. Available from: <https://www.grc.nasa.gov/www/k-12/airplane/isentrop.html>.
10. BENSON, Ben. *Mass Flow Rate Equations* [online]. [N.d.]. [visited on 2025-01-27]. Available from: <https://www.grc.nasa.gov/www/k-12/VirtualAero/BottleRocket/airplane/mchkdirv.html>.
11. CANTWELL, Brian. *AA210A: Fundamentals of Compressible Flow - Course Introduction*. 2020. Presentation slide with a rocket test image on cover.
12. AGRAWAL, Amit. A Comprehensive Review on Gas Flow in Microchannels. *International Journal of Micro-Nano Scale Transport* [online]. 2011, vol. 2, no. 1, pp. 1–40 [visited on 2024-07-09]. ISSN 1759-3093. Available from: <http://doi.org/10.1260/1759-3093.2.1.1>. Number: 1 Publisher: Multi-Science Publishing.
13. LIU, Minghou; ZHANG, Xianfeng; ZHANG, Genxuan; CHEN, Yiliang. Study on micronozzle flow and propulsion performance using DSMC and continuum methods. *Acta Mechanica Sinica* [online]. 2006, vol. 22, no. 5, pp. 409–416 [visited on 2025-02-24]. ISSN 1614-3116. Available from DOI: 10.1007/s10409-006-0020-y.
14. ROBERTSON, S. J. *Investigation of the steady expansion of a gaseous axisymmetric jet into a vacuum Final report* [online]. 1970-01. [visited on 2024-08-22]. Tech. rep., HREC-1490-2. Available from: <https://ntrs.nasa.gov/citations/19700021503>. NTRS Author Affiliations: Lockheed Missiles and Space Co. NTRS Document ID: 19700021503 NTRS Research Center: Legacy CDMS (CDMS).

