

Airfoil Tools

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Your Reynold number range is 1,000,000 to 5,000,000. (info)

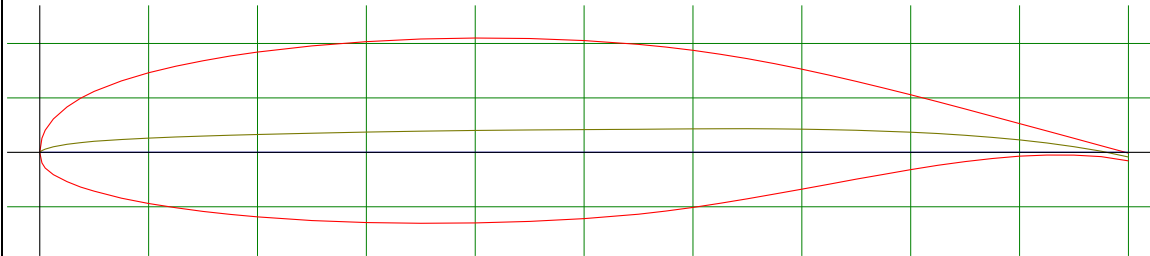
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Search

NASA/LANGLEY LS(1)-0417 (GA(W)-1) AIRFOIL (Is417-il)

NASA/LANGLEY LS(1)-0417 (GA(W)-1) AIRFOIL - NASA/Langley/Whitcomb LS(1)-0417 (GA(W)-1) general aviation airfoil

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Details

(Is417-il) NASA/LANGLEY LS(1)-0417 (GA(W)-1) AIRFOIL
 NASA/Langley/Whitcomb LS(1)-0417 (GA(W)-1) general aviation airfoil
 Max thickness 17% at 40% chord.
 Max camber 2.4% at 65% chord
 Source [UIUC Airfoil Coordinates Database](#)
[Source dat file](#)
 The dat file is in Lednicer format

Dat file

```
NASA/LANGLEY LS(1)-0417 (GA(W)-1)
38.      38.
0.00000  0.00000
.00200   .01300
.00500   .02035
.01000   .03000
```

Parser

No parser warnings

- [Send to airfoil plotter](#)
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Similar airfoils

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- SG6040 [Preview](#) [Details](#)
- NREL's S822 Airfoil [Preview](#) [Details](#)
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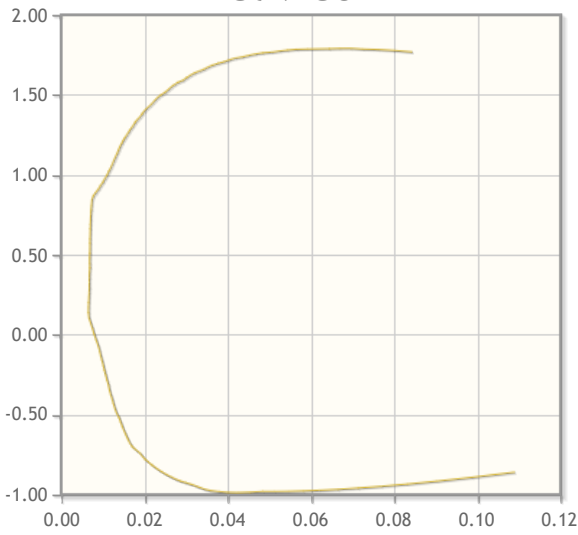


Polars for NASA/LANGLEY LS(1)-0417 (GA(W)-1) AIRFOIL (Is417-il)

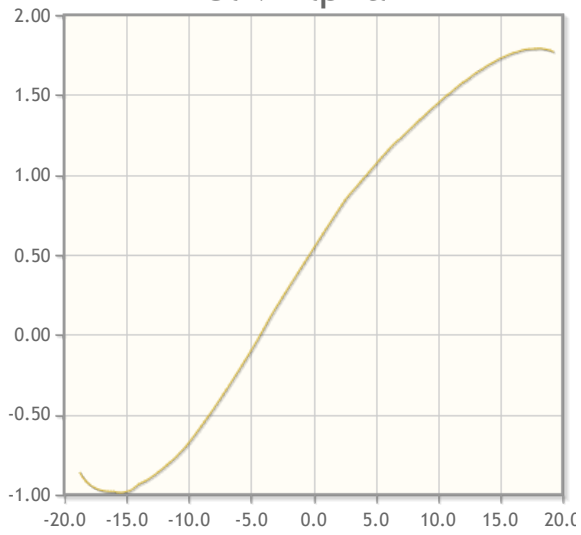
Plot	Airfoil	Reynolds #	Ncrit	Max Cl/Cd	Description	Source
<input type="checkbox"/>	Is417-il	50,000	9	29.7 at $\alpha=8^\circ$	Mach=0 Ncrit=9	Xfoil prediction Details
<input type="checkbox"/>	Is417-il	50,000	5	25 at $\alpha=7.5^\circ$	Mach=0 Ncrit=5	Xfoil prediction Details
<input type="checkbox"/>	Is417-il	100,000	9	42.9 at $\alpha=6.75^\circ$	Mach=0 Ncrit=9	Xfoil prediction Details
<input type="checkbox"/>	Is417-il	100,000	5	41 at $\alpha=5.5^\circ$	Mach=0 Ncrit=5	Xfoil prediction Details
<input type="checkbox"/>	Is417-il	200,000	9	61 at $\alpha=5.25^\circ$	Mach=0 Ncrit=9	Xfoil prediction Details
<input type="checkbox"/>	Is417-il	200,000	5	61.7 at $\alpha=3.75^\circ$	Mach=0 Ncrit=5	Xfoil prediction Details
<input type="checkbox"/>	Is417-il	500,000	9	92.9 at $\alpha=3.75^\circ$	Mach=0 Ncrit=9	Xfoil prediction Details
<input type="checkbox"/>	Is417-il	500,000	5	82.1 at $\alpha=2.25^\circ$	Mach=0 Ncrit=5	Xfoil prediction Details
<input checked="" type="checkbox"/>	Is417-il	1,000,000	9	112.6 at $\alpha=2.5^\circ$	Mach=0 Ncrit=9	Xfoil prediction Details
<input type="checkbox"/>	Is417-il	1,000,000	5	91.3 at $\alpha=1^\circ$	Mach=0 Ncrit=5	Xfoil prediction Details

Global Expansion eBook

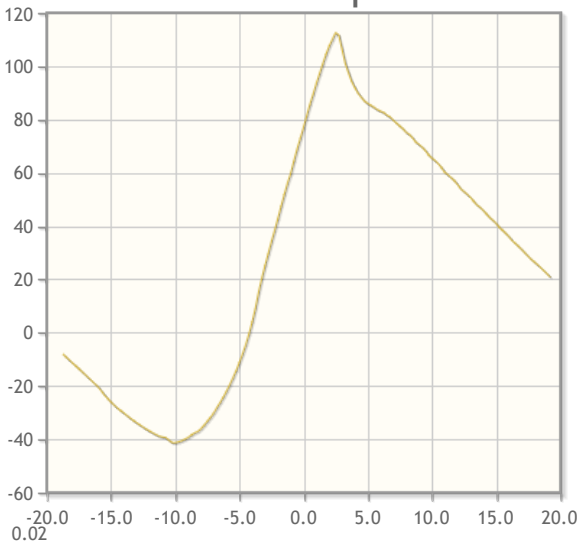
Cl v Cd



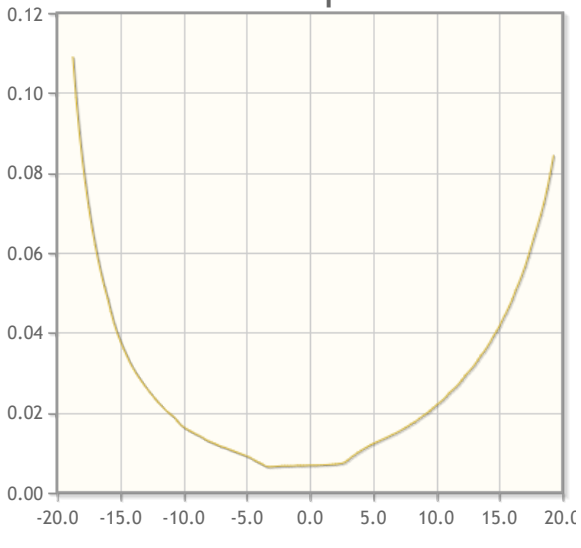
Cl v Alpha



Cl/Cd v Alpha



Cd v Alpha



Cm v Alpha

