

**NIST Special Publication 1018-1**  
**Sixth Edition**

**Fire Dynamics Simulator  
Technical Reference Guide  
Volume 1: Mathematical Model**

Kevin McGrattan  
Simo Hostikka  
Jason Floyd  
Randall McDermott  
Marcos Vanella  
Eric Mueller

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**Fire Safety Research Institute**  
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**NIST**  
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# FDS Developers

The Fire Dynamics Simulator and Smokeview are the products of an international collaborative effort led by the National Institute of Standards and Technology (NIST) and Fire Safety Research Institute, UL Research Institutes. Its developers and contributors are listed below.

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# Preface

This document provides the theoretical basis for the Fire Dynamics Simulator (FDS), following the general framework set forth in the “Standard Guide for Evaluating the Predictive Capability of Deterministic Fire Models,” ASTM E 1355 [1]. This is the first of a four volume set of companion documents, referred to collectively as the FDS Technical Reference Guide [2]. Volumes 2, 3 and 4 describe the model verification, experimental validation, and configuration management, respectively.

A separate document, *Fire Dynamics Simulator, User’s Guide* [3] describes how the FDS software is actually used.



# **Disclaimer**

The US Department of Commerce makes no warranty, expressed or implied, to users of the Fire Dynamics Simulator (FDS), and accepts no responsibility for its use. Users of FDS assume sole responsibility under Federal law for determining the appropriateness of its use in any particular application; for any conclusions drawn from the results of its use; and for any actions taken or not taken as a result of analysis performed using these tools.

Users are warned that FDS is intended for use only by those competent in the fields of fluid dynamics, thermodynamics, heat transfer, combustion, and fire science, and is intended only to supplement the informed judgment of the qualified user. The software package is a computer model that may or may not have predictive capability when applied to a specific set of factual circumstances. Lack of accurate predictions by the model could lead to erroneous conclusions with regard to fire safety. All results should be evaluated by an informed user.

Throughout this document, the mention of computer hardware or commercial software does not constitute endorsement by NIST, nor does it indicate that the products are necessarily those best suited for the intended purpose.



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- The US Nuclear Regulatory Commission Office of Nuclear Regulatory Research funded key validation experiments, the preparation of the FDS manuals, and the development of various sub-models that are of importance in the area of nuclear power plant safety. Special thanks to Mark Salley, Jason Dreisbach, and David Stroup for their efforts and support.
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- The European Union through the FP6 and FP7 research projects FIRE PARADOX, TRANSFEU and FIRE-RESIST.

The following individuals and organizations played a role in the development of the underlying mathematical model of FDS.

- Originally, Ronald Rehm and Howard Baum designed the basic hydrodynamic solver with programming help from Darcy Barnett, Dan Lozier and Hai Tang of the Computing and Applied Mathematics Laboratory at NIST, and Dan Corley of the Building and Fire Research Laboratory (BFRL). Jim Sims of CAML developed the original visualization software.
- The direct Poisson solver (CRAYFISHPAK) was written by Roland Sweet of the National Center for Atmospheric Research (NCAR), Boulder, Colorado.

- Kuldeep Prasad added the multiple-mesh data structures, paving the way for parallel processing.
- Charles Bouldin devised the basic framework of the parallel version of the code.
- William Grosshandler (retired from NIST) and Tom Cleary (currently at NIST) developed an enhancement to the smoke detector activation algorithm, originally conceived by Gunnar Heskstad of Factory Mutual.
- Steve Olenick of Combustion Science and Engineering (CSE) implemented the smoke detector model into FDS.
- William Grosshandler is also the developer of RadCal, a library of subroutines that have been incorporated in FDS to provide the radiative properties of gases and smoke.
- Professor Fred Mowrer, formerly of the University of Maryland, provided a simple model of gas phase extinction to FDS.
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# Chapter 1

## Introduction

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The idea that the dynamics of a fire might be studied numerically dates back to the beginning of the computer age. Indeed, the fundamental conservation equations governing fluid dynamics, heat transfer, and combustion were first written down over a century ago. Despite this, practical mathematical models of fire (as distinct from controlled combustion) are relatively recent due to the inherent complexity of the problem. Indeed, in his brief history of the early days of fire research, Hoyt Hottel noted “A case can be made for fire being, next to the life processes, the most complex of phenomena to understand” [4].

The difficulties revolve about three issues: First, there are an enormous number of possible fire scenarios to consider due to their accidental nature. Second, the physical insight and computing power required to perform all the necessary calculations for most fire scenarios are limited. Any fundamentally based study of fires must consider at least some aspects of bluff body aerodynamics, multi-phase flow, turbulent mixing and combustion, radiative transport, and conjugate heat transfer; all of which are active research areas in their own right. Finally, the “fuel” in most fires was never intended as such. Thus, the mathematical models and the data needed to characterize the degradation of the condensed phase materials that supply the fuel may not be available. Indeed, the mathematical modeling of the physical and chemical transformations of real materials as they burn is still in its infancy.

In order to make progress, the questions that are asked have to be greatly simplified. To begin with, instead of seeking a methodology that can be applied to all fire problems, we begin by looking at a few scenarios that seem to be most amenable to analysis. Hopefully, the methods developed to study these “simple” problems can be generalized over time so that more complex scenarios can be analyzed. Second, we must learn to live with idealized descriptions of fires and approximate solutions to our idealized equations. Finally, the methods should be capable of systematic improvement. As our physical insight and computing power grow more powerful, the methods of analysis can grow with them.

To date, three distinct approaches to the simulation of fires have emerged. Each of these treats the fire as an inherently three dimensional process evolving in time. The first to reach maturity, the “zone” models, describe compartment fires. Each compartment is divided into two spatially homogeneous volumes, a hot upper layer and a cooler lower layer. Mass and energy balances are enforced for each layer, with additional models describing other physical processes appended as differential or algebraic equations as appropriate. Examples of such phenomena include fire plumes; flows through doors, windows and other vents, radiative and convective heat transfer; and solid fuel pyrolysis. Descriptions of the physical and mathematical assumptions behind the zone modeling concept are given in separate papers by Jones [5] and Quintiere [6], who chronicle developments through 1983. Model development since then has progressed to the point where documented and supported software implementing these models are widely available [7].

The relative physical and computational simplicity of the zone models has led to their widespread use in

the analysis of fire scenarios. So long as detailed spatial distributions of physical properties are not required, and the two layer description reasonably approximates reality, these models are quite reliable. However, by their very nature, there is no way to systematically improve them. The rapid growth of computing power and the corresponding maturing of computational fluid dynamics (CFD), has led to the development of CFD based “field” models applied to fire research problems. Virtually all this work is based on the conceptual framework provided by the Reynolds-averaged form of the Navier-Stokes equations (RANS), in particular the  $k - \epsilon$  turbulence model pioneered by Patankar and Spalding [8]. The use of CFD models has allowed the description of fires in complex geometries, and the incorporation of a wide variety of physical phenomena. However, these models have a fundamental limitation for fire applications – the averaging procedure at the root of the model equations.

RANS models were developed as a time-averaged approximation to the conservation equations of fluid dynamics. While the precise nature of the averaging time is not specified, it is clearly long enough to require the introduction of large eddy transport coefficients to describe the unresolved fluxes of mass, momentum and energy. This is the root cause of the smoothed appearance of the results of even the most highly resolved fire simulations. The smallest resolvable length scales are determined by the product of the local velocity and the averaging time rather than the spatial resolution of the underlying computational grid. This property of RANS models is typically exploited in numerical computations by using implicit numerical techniques to take large time steps.

Unfortunately, the evolution of large eddy structures characteristic of most fire plumes is lost with such an approach, as is the prediction of local transient events. It is sometimes argued that the averaging process used to define the equations is an “ensemble average” over many replicates of the same experiment or postulated scenario. However, this is a moot point in fire research since neither experiments nor real scenarios are replicated in the sense required by that interpretation of the equations. The application of “Large Eddy Simulation” (LES) techniques to fire is aimed at extracting greater temporal and spatial fidelity from simulations of fire performed on the more finely meshed grids allowed by ever faster computers.

The phrase LES refers to the description of turbulent mixing of the gaseous fuel and combustion products with the local atmosphere surrounding the fire. This process, which determines the burning rate in most fires and controls the spread of smoke and hot gases, is extremely difficult to predict accurately. This is true not only in fire research but in almost all phenomena involving turbulent fluid motion. The basic idea behind the LES technique is that the eddies that account for most of the mixing are large enough to be calculated with reasonable accuracy from the equations of fluid dynamics. The hope (which must ultimately be justified by comparison to experiments) is that small-scale eddy motion can either be crudely accounted for or ignored.

The equations describing the transport of mass, momentum, and energy by the fire-induced flows must be simplified so that they can be efficiently solved for the fire scenarios of interest. The general equations of fluid dynamics describe a rich variety of physical processes, many of which have nothing to do with fires. Retaining this generality would lead to an enormously complex computational task that would shed very little additional insight on fire dynamics. The simplified equations, developed by Rehm and Baum [9], have been widely adopted by the larger combustion research community, where they are referred to as the “low Mach number” combustion equations. They describe the low speed motion of a gas driven by chemical heat release and buoyancy forces. The phrase low speed refers to gas velocities less than a Mach number of 0.3 (100 m/s); however, it should be noted that current validation of FDS only extends to a Mach number of approximately 0.1. Oran and Boris provide a useful discussion of the technique as applied to various reactive flow regimes in the chapter entitled “Coupled Continuity Equations for Fast and Slow Flows” in Ref. [10]. They comment that “There is generally a heavy price for being able to use a single algorithm for both fast and slow flows, a price that translates into many computer operations per time step often spent in solving multiple and complicated matrix operations.”

The low Mach number equations are solved numerically by dividing the physical space where the fire

is to be simulated into a large number of rectangular cells. Within each cell the gas velocity, temperature, etc., are assumed to be uniform; changing only with time. The accuracy with which the fire dynamics can be simulated depends on the number of cells that can be incorporated into the simulation. This number is ultimately limited by the computing power available. Present day, single processor desktop computers limit the number of such cells to at most a few million. This means that the ratio of largest to smallest eddy length scales that can be resolved by the computation (the “dynamic range” of the simulation) is on the order of 100. Parallel processing can be used to extend this range to some extent, but the range of length scales that need to be accounted for if all relevant fire processes are to be simulated is roughly  $10^4$  to  $10^5$  because combustion processes take place at length scales of 1 mm or less, while the length scales associated with building fires are of the order of tens of meters. The form of the numerical equations discussed below depends on which end of the spectrum one wants to capture directly, and which end is to be ignored or approximated.



## Chapter 2

# Overview of the FDS Model

This chapter presents the governing equations of FDS and an outline of the general solution procedure. Details are included in subsequent chapters. The purpose of this chapter is to highlight aspects of the solution methodology that make it practical for thermally-driven flow simulations, in particular fire. Some of the major features of the model, in its default operation, are:

- Low Mach, large-eddy simulation (LES)
- Explicit, second-order, kinetic-energy-conserving numerics
- Structured, uniform, staggered grid
- Simple immersed boundary method for treatment of flow obstructions
- Generalized “lumped species” method (simplified chemistry using a reaction progress variable)
- Deardorff eddy viscosity subgrid closure
- Constant turbulent Schmidt and Prandtl numbers
- Eddy dissipation concept (fast chemistry) for single-step reaction between fuel and oxidizer
- Gray gas radiation with finite volume solution to the radiation transport equation

The model, however, is not limited to these simple algorithms. For example, the user may specify multiple reactions, finite-rate chemistry, a wide-band radiation model, and a variety of other special features. The more detailed physics incur increased computational cost, and it is incumbent on the user to justify the added expense in terms of improved accuracy for a particular application. The default model options have been selected based on results from a wide variety of full-scale validation experiments [11].

The algorithm outlined below has evolved over roughly three decades. Initially, it was designed to study buoyant plumes in the Boussinesq limit; that is, the fluid was assumed incompressible but included a source term for buoyancy. This approach was based on a long tradition in fire research of modeling smoke movement using dyed salt water introduced into a tank filled with fresh water. Eventually, this approach proved too limiting, but some of the major features of the algorithm, like the low Mach number approximation, were retained.

## 2.1 LES Formalism

The equations for large-eddy simulation (LES) are derived by applying a low-pass filter, parameterized by a width  $\Delta$ , to the transport equations for mass, momentum and energy. For our purposes, it is sufficient to think of the filtered fields in the LES equations as cell means. For example, in 1D the filtered density for a cell of width  $\Delta$  is

$$\bar{\rho}(x,t) = \frac{1}{\Delta} \int_{x-\Delta/2}^{x+\Delta/2} \rho(r,t) dr. \quad (2.1)$$

In FDS, the filter width  $\Delta$  is equivalent to the local cell size  $\delta x$  and is a key parameter in the submodels for the turbulent viscosity and the reaction time scale discussed later. The practice of taking  $\Delta = \delta x$  is called implicit filtering. It is important to appreciate, however, that implicit filtering does not imply dissipative numerics. FDS employs kinetic-energy-conserving central difference schemes for momentum with physically-based closures for the turbulent stress. In what follows, the filter formalism is relaxed (the overline notation is suppressed for clarity) since no explicit filtering operations are performed in the algorithm. A detailed derivation of the formal LES equations is presented in Chapter 4.

## 2.2 Numerical Grid

Because FDS was designed to simulate thermally-driven flows within buildings, the most obvious and simplest numerical grid is rectilinear. In fact, because FDS is a large eddy simulation (LES) model, uniform meshing is preferred, and the only numerical parameters chosen by the end user are the three dimensions of the grid. Once established, it is relatively simple to define rectangular obstructions that define the geometry to the level of resolution determined by the grid. These obstructions “snap” to the underlying grid, a very elementary form of an immersed boundary method (IBM). Unstructured geometry is possible in FDS by virtue of a cutcell method as described in Sec. 12.

### 2.2.1 The Staggered Grid

The governing equations are approximated using second-order accurate finite differences on a collection of uniformly spaced three-dimensional grids. Multiple meshes can be processed in parallel using Message Passing Interface (MPI) libraries. Scalar quantities are assigned to the center of each grid cell; the velocity components at the appropriate cell faces; and vorticity components at cell edges. This is what is commonly referred to as a staggered grid [12, 13]. Its main purpose is to avoid “checker-boarding” in pressure-velocity coupling by naturally representing the pressure cell velocity divergence, a very important thermodynamic quantity in the model. Figure 2.1 displays where in a given grid cell the various flow variables are located.

### 2.2.2 Cell Edges

In the FDS staggered grid arrangement, cell edges are the storage location of both off-diagonal stresses and vorticity. The computational edge variable position environment is described below.

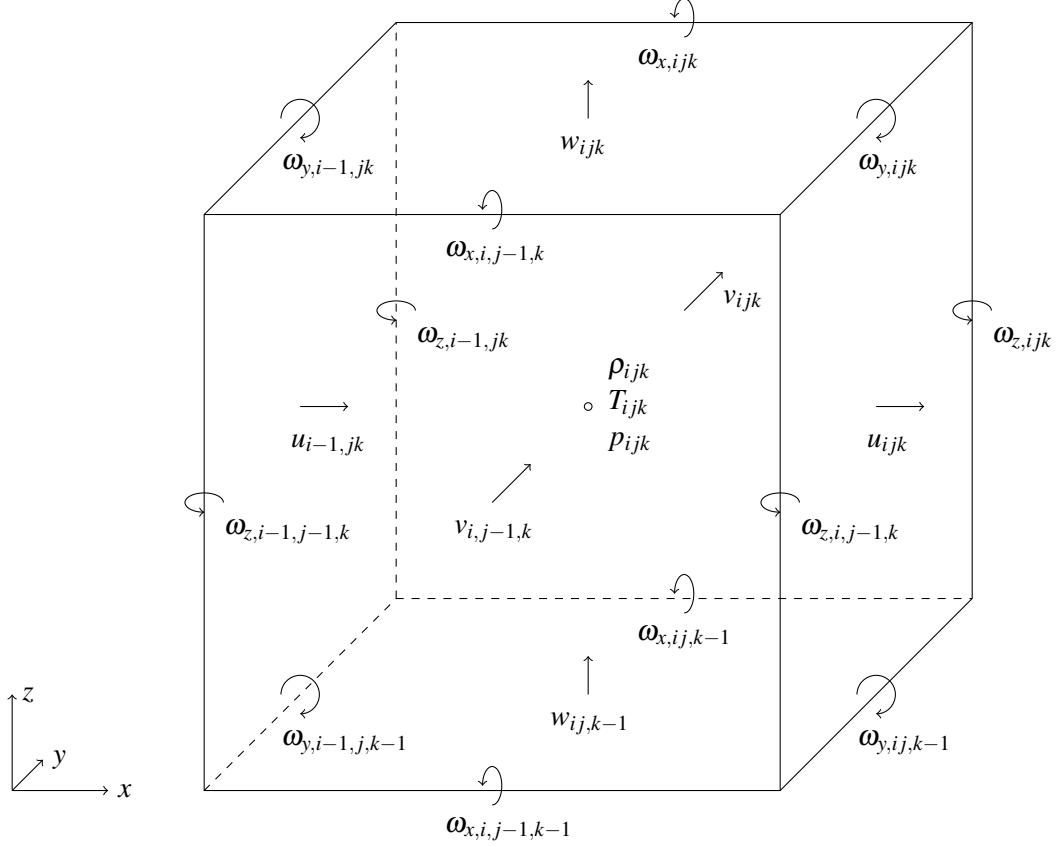


Figure 2.1: Position of flow variables in grid cell  $ijk$ . The arrows indicate the positive direction of the given variable. Scalar variables, such as the density,  $\rho$ , temperature,  $T$ , and pressure,  $p$ , are defined at the cell center. Velocity components,  $\mathbf{u} = (u, v, w)$ , are defined at their respective cell faces, and vorticity components,  $\vec{\omega} = (\omega_x, \omega_y, \omega_z)$ , are located at cell edges.

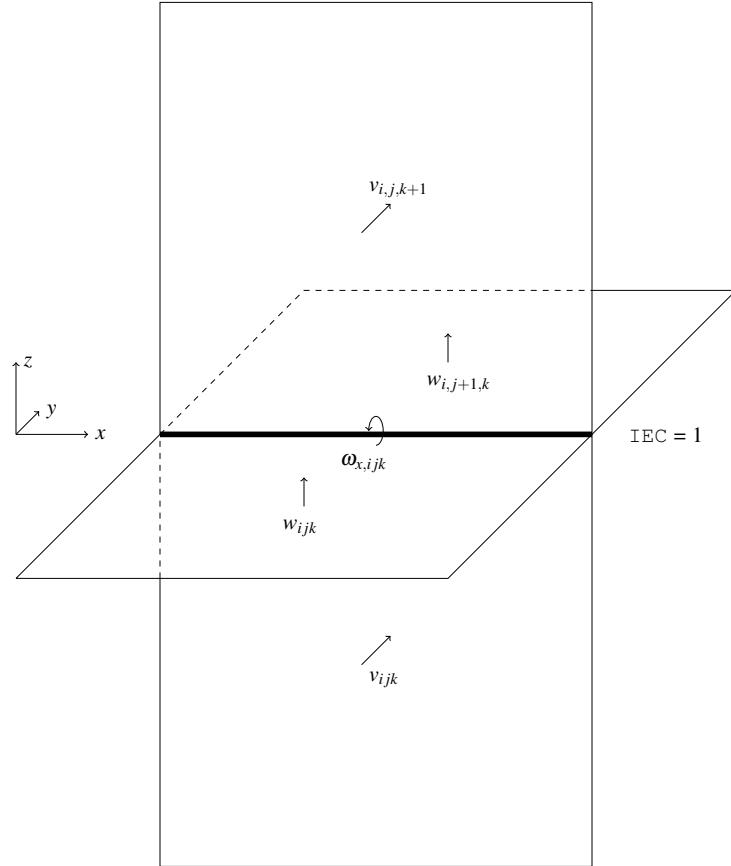
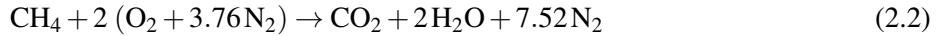


Figure 2.2: Position of flow variables around a computational edge along the  $x$  axis. The Index of the Edge Component (IEC) is 1 in this case. The orientations of the faces (defined by the directions normal to the faces) that connect to the edge are IOR =  $\pm 2, \pm 3$ . The vorticity around the edge is  $\omega_x = \partial w / \partial y - \partial v / \partial z$ .

## 2.3 Mass and Species Transport

The most basic description of the chemistry of fire is a reaction of a hydrocarbon fuel with oxygen that produces carbon dioxide and water vapor. Because fire is a relatively inefficient combustion process involving multiple fuel gases that contain more than just carbon and hydrogen atoms, the number of gas species to keep track of in the simulation is almost limitless. However, to make the simulations tractable, we limit the number of fuels to one, usually, and the number of reactions to just one or two. We also leave open the possibility that the reaction may not proceed for lack of sufficient oxygen in the incoming air stream, as when a fire in a closed compartment extinguishes itself. Even with this simplified approach to the chemistry, we still need to track at least six gas species (Fuel, O<sub>2</sub>, CO<sub>2</sub>, H<sub>2</sub>O, CO, N<sub>2</sub>) plus soot particulate. If we assume a single-step reaction, we do not need to solve explicitly seven transport equations. In fact, we only need to solve two – one for the fuel and one for the products. The air is everything that is neither fuel nor products. However, to ensure realizability of species mass fractions, our strategy is to solve a transport equation for each species mass density and then to obtain the mixture mass density by summation of the species densities.

Whereas the fuel is usually a single gas species, the air and products are what are often referred to as “lumped species”. A lumped species represents a mixture of gas species that transport together (i.e., the lumped species has a single set of transport properties) and react together, and from the point of view of the numerical model, a lumped species can be treated as a single species. In fact, the mass transport equations make no distinction between a single or lumped species. For example, air is a lumped species that consists of nitrogen, oxygen, and trace amounts of water vapor and carbon dioxide. We use the symbols Z<sub>A</sub>, Z<sub>F</sub>, and Z<sub>P</sub> to denote the mass fractions of air, fuel and products. The lumped species mass fractions are linearly related to the primitive species mass fractions, Y<sub>α</sub>; thus, conversion from one to the other is a simple matter of performing a matrix multiplication. For example, the complete combustion of methane:



is expressed as



and the primitive species can be recovered from the lumped species via

$$\begin{bmatrix} 0.77 & 0.00 & 0.73 \\ 0.23 & 0.00 & 0.00 \\ 0.00 & 1.00 & 0.00 \\ 0.00 & 0.00 & 0.15 \\ 0.00 & 0.00 & 0.12 \end{bmatrix} \begin{bmatrix} Z_A \\ Z_F \\ Z_P \end{bmatrix} = \begin{bmatrix} Y_{\text{N}_2} \\ Y_{\text{O}_2} \\ Y_{\text{CH}_4} \\ Y_{\text{CO}_2} \\ Y_{\text{H}_2\text{O}} \end{bmatrix} \quad (2.4)$$

Notice that the columns of the matrix are the mass fractions of the primitive species within a given lumped species.

The transport equation for each of the lumped species has the same form as the transport equation for a single species:

$$\frac{\partial}{\partial t}(\rho Z_\alpha) + \nabla \cdot (\rho Z_\alpha \mathbf{u}) = \nabla \cdot (\rho D_\alpha \nabla Z_\alpha) + \dot{m}'_\alpha'' + \dot{m}_{b,\alpha}''' \quad (2.5)$$

Note that the source term on the right hand side represents the addition of mass from evaporating droplets or other subgrid-scale particles that represent sprinkler and fuel sprays, vegetation, and any other type of small, unresolvable object. These objects are assumed to occupy no volume; thus, they are seen by the governing equations as point sources of mass, momentum, and energy. It is important to note, however, that the evaporated mass species must be one for which an explicit transport equation is solved. For example, water vapor is a product of combustion, but it is also formed by evaporating sprinkler droplets. In cases such

as these, there needs to be an explicit transport equation for water vapor to distinguish between that which is formed by combustion and that which is evaporated from the droplets. Here  $\dot{m}_{b,\alpha}'''$  is the production rate of species  $\alpha$  by evaporating droplets or particles.

The mass density is obtained from  $\rho = \sum(\rho Z)_\alpha$ . The summation of Eq. (2.5) over all  $N_s$  species gives

$$\frac{\partial \rho}{\partial t} + \nabla \cdot (\rho \mathbf{u}) = \dot{m}_b''' \quad (2.6)$$

because  $\sum Z_\alpha = 1$  and  $\sum \dot{m}_\alpha''' = 0$  and  $\sum \dot{m}_{b,\alpha}''' = \dot{m}_b'''$ , by definition, and because it is assumed that  $\sum \rho D_\alpha \nabla Z_\alpha = 0$ . This last assertion is not true, in general. The diffusive flux for the most abundant local species is corrected to enforce the constraint.

### Enforcing Realizability

Realizability of species mass fractions requires  $Y_\alpha \geq 0$  for all  $\alpha$  and  $\sum Y_\alpha = 1$ . Note that this is more restrictive than the boundedness constraint, which simply requires  $0 \leq Y_\alpha \leq 1$ .

If  $(\rho Y)_\alpha$  obeys boundedness,  $(\rho Y)_\alpha \geq 0$ , and we solve  $N_s$  species equations obtaining the density via  $\rho = \sum_{\alpha=1}^{N_s} (\rho Y)_\alpha$ , then mass fractions obtained by  $Y_\alpha = (\rho Y)_\alpha / \rho$  are guaranteed to be realizable (for  $\rho > 0$ ). Thus, we have reduced the realizability problem to the “easier” problem of boundedness for  $(\rho Y)_\alpha$ . Details of the scalar boundedness correction are discussed in Appendix G.

With this approach we must take care to ensure  $\sum \rho D_\alpha \nabla Z_\alpha = 0$ . Our strategy is to absorb any errors in diffusive transport into the most abundant species *locally*. That is, for a given cell face we set  $\rho D_m \nabla Z_m = -\sum_{\alpha \neq m} \rho D_\alpha \nabla Z_\alpha$ , where  $m$  is the most abundant species adjacent to that face. Note that since FDS is typically used as an LES code mass transport by molecular diffusion may be two or three orders of magnitude less than turbulent transport, which uses the same turbulent diffusion coefficient for all species. Therefore, the errors in summation of the diffusive fluxes tend to be small.

## 2.4 Low Mach Number Approximation

For low speed applications like fire, Rehm and Baum [9] observed that the spatially and temporally resolved pressure,  $p$ , can be decomposed into a “background” pressure,  $\bar{p}(z,t)$ , plus a perturbation,  $\tilde{p}(x,y,z,t)$ , with only the background pressure retained in the equation of state (ideal gas law):

$$\bar{p} = \rho T R \sum_\alpha \frac{Z_\alpha}{W_\alpha} \equiv \frac{\rho R T}{\bar{W}} \quad (2.7)$$

Note that  $z$  is the spatial coordinate in the direction of gravity; thus, the stratification of the atmosphere is included in the background pressure. The perturbation,  $\tilde{p}$ , drives the fluid motion. This approximation has a number of consequences. First, building compartments connected via a heating, ventilation, and air conditioning (HVAC) system can each maintain individual background pressures. The air flows between compartments can be described in terms of the differences in the background pressures, eliminating the need to solve detailed flow equations within the ventilation ducts.

The second consequence of the low Mach number approximation is that the internal energy,  $e$ , and enthalpy,  $h$ , may be related in terms of the thermodynamic (background) pressure:  $h = e + \bar{p}/\rho$ . The energy conservation equation may then be written in terms of the *sensible enthalpy*,  $h_s$ :

$$\frac{\partial}{\partial t} (\rho h_s) + \nabla \cdot (\rho h_s \mathbf{u}) = \frac{D\bar{p}}{Dt} + \dot{q}''' + \dot{q}_b''' - \nabla \cdot \dot{\mathbf{q}}'' \quad (2.8)$$

The term  $\dot{q}'''$  is the heat release rate per unit volume from a chemical reaction. The term  $\dot{q}_b'''$  is the energy transferred to subgrid-scale droplets and particles. The term  $\dot{\mathbf{q}}''$  represents the conductive, diffusive, and radiative heat fluxes:

$$\dot{\mathbf{q}}'' = -k\nabla T - \sum_{\alpha} h_{s,\alpha} \rho D_{\alpha} \nabla Z_{\alpha} + \dot{\mathbf{q}}_r'' \quad (2.9)$$

where  $k$  is the thermal conductivity,  $D_{\alpha}$  is the diffusivity of species  $\alpha$ , and  $\dot{\mathbf{q}}_r''$  is the radiative flux.

Eq. (2.8) is not solved explicitly. Instead, the velocity divergence is factored out as follows:

$$\nabla \cdot \mathbf{u} = \frac{1}{\rho h_s} \left[ \frac{D}{Dt} (\bar{p} - \rho h_s) + \dot{q}''' + \dot{q}_b''' - \nabla \cdot \dot{\mathbf{q}}'' \right] \quad (2.10)$$

The hydrodynamics solver guarantees that Eq. (2.10) is satisfied. It follows that Eq. (2.8) is also satisfied (energy is conserved).

Expanding the material derivatives on the right hand side of Eq. (2.10) produces a fairly complicated expression for the divergence that includes the source and diffusion terms from the mass, species, and energy conservation equations. Its importance to the overall algorithm is that it can be computed using only the thermodynamic variables  $\rho$ ,  $Z_{\alpha}$ , and  $\bar{p}$ . As will be shown below, the way to advance the flow velocity in time is to first estimate the thermodynamic variables at the next time step, compute the divergence, and then solve an equation for the pressure that will guarantee that the divergence of the updated velocity is identical to that computed solely from the thermodynamic variables.

## 2.5 Momentum Transport

Noting the vector identity  $(\mathbf{u} \cdot \nabla) \mathbf{u} = \nabla |\mathbf{u}|^2 / 2 - \mathbf{u} \times \boldsymbol{\omega}$ , and defining the stagnation energy per unit mass,  $H \equiv |\mathbf{u}|^2 / 2 + \tilde{p} / \rho$ , the momentum equation can be written (see Chapter 4 for a detailed derivation)

$$\frac{\partial \mathbf{u}}{\partial t} - \mathbf{u} \times \boldsymbol{\omega} + \nabla H - \tilde{p} \nabla (1/\rho) = \frac{1}{\rho} \left[ (\rho - \rho_0) \mathbf{g} + \mathbf{f}_b + \nabla \cdot \boldsymbol{\tau} \right] \quad (2.11)$$

The term  $\mathbf{f}_b$  represents the drag force exerted by the subgrid-scale particles and droplets. The viscous stress,  $\boldsymbol{\tau}$ , is closed via gradient diffusion with the turbulent viscosity obtained from the Deardorff eddy viscosity model [14, 15]. It is convenient to write Eq. (2.11) in the form:

$$\frac{\partial \mathbf{u}}{\partial t} + \mathbf{F} + \nabla H = 0 \quad (2.12)$$

so that a Poisson equation for the pressure can be derived by taking its divergence:

$$\nabla^2 H = - \left[ \frac{\partial}{\partial t} (\nabla \cdot \mathbf{u}) + \nabla \cdot \mathbf{F} \right] \quad (2.13)$$

Note the appearance of the time derivative of the divergence. This is an important feature of the time marching scheme. Note also that the right hand side of the Poisson equation retains a term that includes the perturbation pressure,  $\tilde{p} \nabla (1/\rho)$ . This term accounts for the baroclinic torque. It is included on the right hand side of the Poisson equation by using its value from the previous time step. This approximation allows us to solve a separable form of the Poisson equation, for which there are fast, direct solvers that are optimized for uniform grids [16].

## 2.6 Combustion and Radiation

FDS is described as a “fire model” because it incorporates source terms and boundary conditions that describe the turbulent combustion of gaseous fuel and oxygen, the transport of thermal radiation through hot, soot-laden gases, the thermal decomposition of real materials, the activation of sprinklers and smoke detectors, the transport of water and liquid fuel droplets, and a variety of other features that describe fires inside and outside of buildings.

Combustion and radiation are introduced into the governing equations via the source terms,  $\dot{q}'''$  and  $\dot{q}_r'''$ , in the energy transport equation. Since the energy equation is not solved explicitly, these terms find their way into the expression for the divergence.

### 2.6.1 Combustion

For most applications, FDS uses a combustion model based on the mixing-limited, infinitely fast reaction of lumped species. Lumped species are reacting scalar quantities that represent a mixture of species. For example, air is a lumped species which is a mixture of nitrogen, oxygen, water vapor, and carbon dioxide. The reaction of fuel and oxygen is not necessarily instantaneous and complete, and there are several optional schemes that are designed to predict the extent of combustion in under-ventilated spaces.

For an infinitely-fast reaction, reactant species in a given grid cell are converted to product species at a rate determined by a characteristic mixing time,  $\tau_{\text{mix}}$ . The heat release rate per unit volume is defined by summing the lumped species mass production rates times their respective heats of formation

$$\dot{q}''' = - \sum_{\alpha} \dot{m}_{\alpha}''' \Delta h_{f,\alpha} \quad (2.14)$$

Details of  $\tau_{\text{mix}}$  and  $\dot{m}_{\alpha}'''$  are discussed in Chapter 5.

### 2.6.2 Radiation

The net contribution from thermal radiation in the energy equation is defined by:

$$\dot{q}_r''' \equiv -\nabla \cdot \dot{\mathbf{q}}_r''(\mathbf{x}) = \kappa(\mathbf{x}) [U(\mathbf{x}) - 4\pi I_b(\mathbf{x})] \quad ; \quad U(\mathbf{x}) = \int_{4\pi} I(\mathbf{x}, \mathbf{s}') d\mathbf{s}' \quad (2.15)$$

where  $\kappa(\mathbf{x})$  is the absorption coefficient,  $I_b(\mathbf{x})$  is the source term, and  $I(\mathbf{x}, \mathbf{s})$  is the solution of the radiation transport equation (RTE) for a non-scattering gray gas:

$$\mathbf{s} \cdot \nabla I(\mathbf{x}, \mathbf{s}) = \kappa(\mathbf{x}) [I_b(\mathbf{x}) - I(\mathbf{x}, \mathbf{s})] \quad (2.16)$$

In practical simulations, the spectral dependence of  $I$ ,  $I_b$ , and  $\kappa$  cannot be resolved accurately, nor do we have reliable data for non-ideal fuels typical of real fires. While FDS does have an option to divide the radiation spectrum into a relatively small number of bands and solve a separate RTE for each band, it is usually not necessary because in real fires, soot is the dominant source and sink of thermal radiation and is not particularly sensitive to wavelength. The mean absorption coefficient,  $\kappa$ , is a function of species composition and temperature. Its values are obtained from a narrow-band model called RadCal [17].

The source term,  $I_b$ , requires special treatment because of the limited resolution of the underlying numerical grid in the vicinity of flames. In large scale fire simulations, grid cells are typically on the order of tens of centimeters. Flame sheets cannot be resolved, meaning that the computed cell-average temperature can be significantly lower than temperatures one would expect to find in the reacting flame. Consequently, the

source term is approximated in grid cells where fuel and oxygen react. Elsewhere, the subgrid temperature field is homogeneous and the source term can be computed directly:

$$\kappa I_b = \begin{cases} \kappa \sigma T^4 / \pi & \text{Outside flame zone, } \dot{q}''' = 0 \\ C \kappa \sigma T^4 / \pi & \text{Inside flame zone, } \dot{q}''' > 0 \end{cases} \quad (2.17)$$

The constant  $C$  is computed at each time step so that the volume integral of Eq. (2.15) over the entire flaming region is approximately equal to the volume integral of  $\chi_r \dot{q}'''$  over that same region. Here,  $\chi_r$  is an empirical estimate of the *global* fraction of that energy emitted as thermal radiation. Typically, a sooty fire radiates approximately one-third of the total combustion energy.

The radiation equation is solved using a technique similar to a finite volume method for convective transport, thus the name given to it is the Finite Volume Method (FVM). Using approximately 100 discrete angles which are updated over multiple time steps, the finite volume solver requires about 20% of the total CPU time of a calculation, a modest cost given the complexity of radiation heat transfer.

Water droplets can absorb and scatter thermal radiation. This is important in scenarios involving water mist suppression systems, but also plays a role in all sprinkler cases. The absorption and scattering coefficients are based on Mie theory. The scattering from the gaseous species and soot is considered negligible and is not included in the model.

## 2.7 Solution Procedure

In a given grid cell at the  $n$ th time step, we have the density,  $\rho^n$ , lumped species mass fractions,  $Z_\alpha^n$ , velocity vector,  $\mathbf{u}^n$ , and the Bernoulli integral,  $H^n$ . In addition, for each compartment in the computational domain, we have a background pressure,  $\bar{p}^n$ . The temperature is found from the equation of state. These variables are advanced in time using an explicit second-order predictor/corrector scheme. The basic procedure is as follows:

### Predictor

1. Estimate  $\rho$ ,  $Z_\alpha$ , and  $\bar{p}$  at the next time step with an explicit Euler step. The species mass density is estimated by

$$\frac{(\rho Z)_\alpha^* - \rho^n Z_\alpha^n}{\delta t} + \nabla \cdot \rho^n Z_\alpha^n \mathbf{u}^n = \nabla \cdot (\rho^n D_\alpha^n \nabla Z_\alpha^n) + (\dot{m}_\alpha''' + \dot{m}_{b,\alpha}''')^n \quad (2.18)$$

The asterisk denotes a first order accurate estimate at the next time step. Note that the mass source terms are computed in the previous time step using the corrected value of the density and species mass fractions.

2. Compute the density from  $\rho^* = \sum_\alpha (\rho Z_\alpha)^*$  and mass fractions from  $Z_\alpha^* = (\rho Z)_\alpha^* / \rho^*$ .
3. Compute the temperature,  $T^*$ , from the equation of state.
4. Compute the divergence,  $(\nabla \cdot \mathbf{u})^*$ , from Eq. (2.10) using the estimated thermodynamic quantities. Note that we use the parentheses to emphasize that an estimate of the velocity field,  $\mathbf{u}^*$ , at the next time step has not been computed yet, only its divergence.
5. Solve the Poisson equation for the pressure term:

$$\nabla^2 H^n = - \frac{(\nabla \cdot \mathbf{u})^* - \nabla \cdot \mathbf{u}^n}{\delta t} - \nabla \cdot \mathbf{F}^n \quad (2.19)$$

6. Estimate the velocity at the next time step.

$$\frac{\mathbf{u}^* - \mathbf{u}^n}{\delta t} + \mathbf{F}^n + \nabla H^n = 0 \quad (2.20)$$

Note that this procedure guarantees that the divergence of the estimated velocity field,  $\nabla \cdot \mathbf{u}^*$ , is identically equal to the divergence that is derived from the estimated thermodynamic quantities,  $(\nabla \cdot \mathbf{u})^*$ , in Step 4.

7. Check that the time step,  $\delta t$ , satisfies the stability condition (see Sec. 4.5). If the time step is too large, it is reduced so that it satisfies the stability constraint and the procedure returns to the beginning of the time step. If the stability criterion is satisfied, the procedure continues to the corrector step.

## Corrector

1. Correct the transported species mass densities at the next time step.

$$\frac{(\rho Z_\alpha)^{n+1} - \frac{1}{2}(\rho^n Z_\alpha^n + \rho^* Z_\alpha^*)}{\delta t / 2} + \nabla \cdot \rho^* Z_\alpha^* \mathbf{u}^* = \nabla \cdot (\rho^* D_\alpha^* \nabla Z_\alpha^*) + (\dot{m}_\alpha''' + \dot{m}_{b,\alpha}''')^n \quad (2.21)$$

The background pressure is corrected similarly. Note that the mass source terms are the same as those added in the predictor step. They are only computed once per time step.

2. Compute the density  $\rho^{n+1} = \sum_\alpha (\rho Z_\alpha)^{n+1}$  and mass fractions  $Z_\alpha^{n+1} = (\rho Z_\alpha)^{n+1} / \rho^{n+1}$ .
3. Compute the temperature,  $T^{n+1}$ , from the equation of state.
4. *Time splitting for mass source terms.* After the corrector step for the transport scheme, source terms are computed and stored. The source terms are evaluated using the results from the corrected scalar transport scheme. The source terms include the heat release rate per unit volume,  $\dot{q}'''$ , the net absorption/emittance of thermal radiation,  $\nabla \cdot \dot{\mathbf{q}}''$ , and the mass species source terms,  $\dot{m}_\alpha'''$ . In addition, the terms in the divergence expression, Eq. (B.7), involving the source terms are computed and stored. All of these quantities are computed at this point in the time step and applied in both the predictor and corrector steps of the following time step.
5. Compute the divergence,  $(\nabla \cdot \mathbf{u})^{n+1}$ , from the corrected thermodynamic quantities.
6. Compute the pressure using the estimated quantities.

$$\nabla^2 H^* = - \left[ \frac{(\nabla \cdot \mathbf{u})^{n+1} - \frac{1}{2}(\nabla \cdot \mathbf{u}^* + \nabla \cdot \mathbf{u}^n)}{\delta t / 2} \right] - \nabla \cdot \mathbf{F}^* \quad (2.22)$$

7. Correct the velocity at the next time step.

$$\frac{\mathbf{u}^{n+1} - \frac{1}{2}(\mathbf{u}^* + \mathbf{u}^n)}{\delta t / 2} + \mathbf{F}^* + \nabla H^* = 0 \quad (2.23)$$

Note again that the divergence of the corrected velocity field is identically equal to the divergence that was computed in Step 5.

## Chapter 3

# Mass, Species, and Enthalpy Transport

This chapter describes in detail the equation of state in the low Mach number limit, the finite difference approximation of the mass and species conservation equations, and the role of the flow divergence as a surrogate for the enthalpy transport equation. Due to the use of the low Mach number approximation, the energy conservation equation is not solved explicitly but rather is defined implicitly via the divergence of the flow field, which contains the combustion and radiation source terms.

### 3.1 The Equation of State

A distinguishing feature of a CFD model is the regime of flow speeds (relative to the speed of sound) for which it is designed. High speed flow codes involve compressibility effects and shock waves. Low speed solvers, however, explicitly eliminate compressibility effects that give rise to acoustic (sound) waves. The Navier-Stokes equations describe the propagation of information at speeds comparable to that of the fluid flow (for fire, approximately 10 m/s), but also at speeds comparable to that of sound waves (for still air, 300 m/s). Solving a discretized form of these equations would require extremely small time steps in order to account for information traveling at the speed of sound, making practical simulations difficult.

Following the work of Rehm and Baum [9], an approximation to the equation of state is made by decomposing the pressure into a “background” component and a perturbation. It is assumed that the background component of pressure can differ from compartment to compartment. If a volume within the computational domain is isolated from other volumes, except via leak paths or ventilation ducts, it is referred to as a “pressure zone” and assigned its own background pressure. The pressure field within the  $m$ th zone, for example, is a linear combination of its background component and the flow-induced perturbation:

$$p(\mathbf{x}, t) = \bar{p}_m(z, t) + \tilde{p}(\mathbf{x}, t) \quad (3.1)$$

Note that the background pressure is a function of  $z$ , the vertical spatial coordinate, and the time,  $t$ . For most compartment fire applications,  $\bar{p}_m$  changes very little with height or time. However, for scenarios where a fire increases the pressure in a closed compartment, or where the HVAC system affects the pressure, or when the height of the domain is significant,  $\bar{p}_m$  takes these effects into account [18]. The ambient pressure field is denoted  $\bar{p}_0(z)$ . Note that the subscript 0 denotes the exterior of the computational domain, not time 0. This is the assumed atmospheric pressure stratification that serves as both the initial and boundary condition for the governing equations.

The purpose of decomposing the pressure is that for low Mach number flows, it can be assumed that the temperature and density are inversely proportional, and thus the equation of state (in the  $m$ th pressure zone) can be approximated as

$$\bar{p}_m = \rho T R \sum_{\alpha} \frac{Z_{\alpha}}{W_{\alpha}} = \frac{\rho T R}{\bar{W}} \quad (3.2)$$

Recall from Sec. 2.3 that  $Z_\alpha$  is the mass fraction of lumped species  $\alpha$ . The pressure,  $p$ , in the state and energy equations is replaced by the background pressure  $\bar{p}_m$  to filter out sound waves that travel at speeds that are much faster than typical flow speeds expected in fire applications. The low Mach number assumption serves two purposes. First, the filtering of acoustic waves means that the time step in the numerical algorithm is bound only by the flow speed as opposed to the speed of sound, and second, the modified state equation leads to a reduction in the number of dependent variables in the system of equations by one. The energy equation (2.8) is not explicitly solved; rather, its source terms are included in the expression for the flow divergence, to be discussed later in the chapter. When the velocity field satisfies the specified thermodynamic divergence, the conservative form of the sensible enthalpy equation is satisfied by construction.

The stratification of the atmosphere is derived from the relation

$$\frac{d\bar{p}_0}{dz} = -\rho_0(z) g \quad (3.3)$$

where  $\rho_0$  is the background density and  $g$  is the acceleration due to gravity. Using Eq. (3.2), the background pressure can be written as a function of the background temperature,  $T_0(z)$ ,

$$\bar{p}_0(z) = p_\infty \exp \left( - \int_{z_\infty}^z \frac{\bar{W} g}{R T_0(z')} dz' \right) \quad (3.4)$$

where the subscript infinity generally refers to the ground. A linear temperature stratification of the atmosphere may be specified by the user such that  $T_0(z) = T_\infty + \Gamma z$  where  $T_\infty$  is the temperature at the ground and  $\Gamma$  is the lapse rate (e.g.,  $\Gamma = -0.0098 \text{ K/m}$  is the *adiabatic lapse rate*). In this case  $\bar{p}_0$  and  $\rho_0$  are derived from Eqs. (3.4) and (3.2), respectively. It can then be shown that for  $\Gamma \neq 0$  the pressure stratification becomes

$$\bar{p}_0(z) = p_\infty \left( \frac{T_0(z)}{T_\infty} \right)^{\bar{W} g / R \Gamma} \quad (3.5)$$

## 3.2 Mass and Species Transport

The species transport equations are solved using a predictor-corrector scheme. Advection terms are written in flux divergence (conservative) form. In the predictor step, the mass density in cell  $ijk$  at time level  $n+1$  is estimated based on information at the  $n$ th level:

$$\frac{(\rho Z_\alpha)_{ijk}^* - (\rho Z_\alpha)_{ijk}^n}{\delta t} + \nabla \cdot (\bar{\rho Z}_\alpha^{\text{FL}} \mathbf{u})_{ijk}^n = \nabla \cdot (\rho D_\alpha \nabla Z_\alpha)_{ijk}^n + (\dot{m}_\alpha''' + \dot{m}_{\text{b},\alpha}''')_{ijk}^n \quad (3.6)$$

The quantity  $\bar{\rho Z}_\alpha^{\text{FL}}$  indicates a *flux limiter* applied to the cell face value, as discussed below in Section 3.2.1. The mass source terms due to chemistry, evaporation, or pyrolysis are computed at the end of the previous time step and used in both the predictor and corrector steps. The mean chemical source term,  $\dot{m}_\alpha'''$ , is discussed in Chapter 5. The bulk subgrid source term,  $\dot{m}_{\text{b},\alpha}'''$ , is discussed in Chapters 7 and 9 on solid phase pyrolysis and Lagrangian particles, respectively.

In DNS mode, the molecular diffusivity is based on mixture-averaged binary Fickian diffusion. In LES mode (default) the diffusivity is taken from the molecular and turbulent viscosities divided by the turbulent Schmidt number. That is, to save cost we approximate the molecular plus turbulent diffusivity by  $(\mu + \mu_t)/Sc_t$ . The turbulent Schmidt number is constant with default value  $Sc_t = 0.5$ . The model for the turbulent viscosity  $\mu_t$  is discussed in Section 4.2. Optionally, by setting `SIMULATION_MODE='LES'` on `MISC`, the molecular and turbulent transport coefficients are treated separately,  $\rho D_\alpha + \mu_t/Sc_t$  (at added cost). The same applies for the thermal diffusivity.

The corrector step is as follows:

$$\frac{(\rho Z_\alpha)_{ijk}^{n+1} - \frac{1}{2} \left[ (\rho Z_\alpha)_{ijk}^n + (\rho Z_\alpha)_{ijk}^* \right]}{\frac{1}{2} \delta t} + \nabla \cdot (\overline{\rho Z_\alpha}^{\text{FL}} \mathbf{u})_{ijk}^* = \nabla \cdot (\rho D_\alpha \nabla Z_\alpha)_{ijk}^* + (\dot{m}_\alpha''' + \dot{m}_{\text{b},\alpha}'')_{ijk}^n \quad (3.7)$$

### 3.2.1 Flux Limiters

A *flux limiter* is an interpolation scheme for defining mass fluxes at cell faces. Simple linear interpolation of the cell-centered scalar variables to the cell face would result in a central difference scheme. Such purely centered schemes are known to generate intolerable levels of dispersion error (spurious wiggles) leading to unphysical results such as negative densities or mass fractions outside the range of [0,1]. To address this issue, FDS relies on two schemes: a *flux limiter* (discussed below) that handles the bulk of the problem, and a *flux correction* (see Appendix G) that adds the minimum amount of numerical diffusion to maintain boundedness.

For uniform flow velocity, a fundamental property of the exact solution to the equations governing scalar transport is that the total variation of the scalar field (the sum of the absolute values of the scalar differences between neighboring cells) is either preserved or diminished (never increased). In other words, no new extrema are created. Numerical schemes which preserve this property are referred to as total variation diminishing (TVD) schemes. The practical importance of using a TVD scheme for fire modelling is that such a scheme is able to accurately track coherent vortex structure in turbulent flames and does not develop spurious reaction zones.

FDS employs two second-order TVD schemes as options for scalar transport: Superbee and CHARM. Superbee [19] is recommended for LES because it more accurately preserves the scalar variance for coarse grid solutions that are not expected to be smooth. Due to the gradient steepening applied in Superbee, however, the convergence degrades at small grid spacing for smooth solutions (the method will revert to a stair-step pattern instead of the exact solution). CHARM [20], though slightly more dissipative than Superbee, is convergent, and is therefore the better choice for DNS calculations where the flame front is well resolved.

To illustrate how flux limiters are applied to the scalar transport equations, below we discretize the advection terms in Eq. (2.5) in one dimension:

$$\frac{(\rho Z)_i^* - (\rho Z)_i^n}{\delta t} + \frac{\overline{\rho Z}_{i+\frac{1}{2}}^{\text{FL}} u_{i+\frac{1}{2}} - \overline{\rho Z}_{i-\frac{1}{2}}^{\text{FL}} u_{i-\frac{1}{2}}}{\delta x} = \dots \quad (3.8)$$

Note that the  $\pm \frac{1}{2}$  suffixes indicates a face value for a particular cell  $i$ . A flux-limited scalar value (density in this case) premultiplies the staggered, face-centered velocity to form the scalar advective flux. Recall that these velocity values are primitive variables in the calculation—they are *not* interpolated. Consider face  $i + \frac{1}{2}$  between cells  $i$  and  $i + 1$  and let  $\phi$  denote a general scalar variable, like  $\rho Z_\alpha$ . The local (loc) and upstream (up) data variations are

$$\delta \phi_{\text{loc}} = \phi_{i+1} - \phi_i \quad (3.9)$$

$$\delta \phi_{\text{up}} = \begin{cases} \phi_i - \phi_{i-1} & \text{if } u_i > 0 \\ \phi_{i+2} - \phi_{i+1} & \text{if } u_i < 0 \end{cases} \quad (3.10)$$

The limiter function  $B(r)$  depends on the upstream-to-local data ratio,  $r = \delta \phi_{\text{up}} / \delta \phi_{\text{loc}}$ . In FDS, options for the limiter function include [21]:

Flux Limiter	$B(r)$
Central Difference	1
Godunov	0
MINMOD	$\max(0, \min(1, r))$
Superbee [19] (LES default)	$\max(0, \min(2r, 1), \min(r, 2))$
CHARM [20] (DNS default)	$s(3s+1)/(s+1)^2; s = 1/r$
MP5 [22]	see below

For the Central Difference, Godunov, MINMOD, and Superbee limiters, the scalar face value is found from

$$\bar{\phi}_{i+1/2}^{\text{FL}} = \begin{cases} \phi_i + B(r) \frac{1}{2} \delta\phi_{\text{loc}} & \text{if } u_i > 0 \\ \phi_{i+1} - B(r) \frac{1}{2} \delta\phi_{\text{loc}} & \text{if } u_i < 0 \end{cases} \quad (3.11)$$

For CHARM, the face value is given by [23]

$$\bar{\phi}_{i+1/2}^{\text{FL}} = \begin{cases} \phi_i + B(r) \frac{1}{2} \delta\phi_{\text{up}} & \text{if } u_i > 0 \\ \phi_{i+1} - B(r) \frac{1}{2} \delta\phi_{\text{up}} & \text{if } u_i < 0 \end{cases} \quad (3.12)$$

The MP5 scheme of Suresh and Huynh [22] is based on the keen observation that three points cannot distinguish between extrema and discontinuities. The functional form of the limiter is not as simple as the three-point schemes described above, so we refer the reader to the original paper or the FDS source code for details. But the basic idea behind the method is to use a five-point stencil, three upwind and two downwind, to reconstruct the cell face value, considering both accuracy and monotonicity-preserving constraints. An additional benefit of the MP5 scheme is that it was designed specifically with strong stability-preserving (SSP) Runge-Kutta time discretizations in mind. The predictor-corrector scheme used by FDS is similar to the second-order SSP scheme described in [24].

### Notes on Implementation

In practice, we set  $r = 0$  initially and only compute  $r$  if the denominator is not zero. Note that for  $\delta\phi_{\text{loc}} = 0$ , it does not matter which limiter is used: all the limiters yield the same scalar face value. For CHARM, we set both  $r = 0$  and  $B = 0$  initially and only compute  $B$  if  $r > 0$  (this requires data variations to have the same sign). Otherwise, CHARM reduces to Godunov's scheme.

The Central Difference, Godunov, and MINMOD limiters are included for completeness, debugging, and educational purposes. These schemes have little utility for typical FDS applications.

### 3.2.2 Time Splitting for Mass Source Terms

Following the corrector step of the transport scheme, source terms are computed for the next time step. The source terms are typically related to particle evaporation or combustion, and these processes are computed at the end of the time step. In the case of combustion, the total mass of a grid cell is not changed; rather the species mass fractions change. The mean chemical source term,  $\dot{m}_{\alpha}'''$ , is discussed in Chapter 5. The bulk subgrid source term,  $\dot{m}_{b,\alpha}'''$ , is discussed in Chapters 7 and 9 on solid phase pyrolysis and Lagrangian particles, respectively.

### 3.2.3 Boundary Conditions for Temperature, Species Mass Fraction, and Density

The gas temperature, species mass fractions, and density are computed at the center of each grid cell. At an exterior boundary, or at the boundary of an interior obstruction, these values must be computed at the face

of the cell that falls at the boundary interface. In general, the temperature at the boundary,  $T_w$ , is computed first, followed by species mass fractions,  $Z_{\alpha,w}$ , followed by density,  $\rho_w$ . The density is typically determined from the equation of state:

$$\rho_w = \frac{\bar{p}_m}{R T_w \sum_{\alpha} (Z_{\alpha,w} / W_{\alpha})} \quad (3.13)$$

Here,  $\bar{p}_m$  denotes the background pressure of the gas phase region.

When necessary, the boundary value is linearly extrapolated one half of a grid cell into the “ghost” cell for use by the gas phase solver. In the sections below, the value at the center of the gas phase cell adjacent to the boundary is denoted with the subscript “g” (for “gas phase”, *not* “ghost”), and the value at the boundary by “w” (for “wall”).

## Solid Boundaries

At a solid boundary, the surface temperature,  $T_w$ , is either specified or computed as described in Chapter 7. For an LES calculation, the convective heat flux at the surface is determined via an empirical heat transfer coefficient,  $h$ , and the convective heat flux at the boundary is written:

$$k \frac{T_g - T_w}{\delta n/2} = h (T_g - T_w) \quad (3.14)$$

where  $\delta n/2$  is the distance between the surface and the center of the adjacent gas phase cell. The convective heat transfer coefficient,  $h$ , is described in Sec. 7.1.2. For a DNS calculation, the convective heat transfer is determined directly from the computed or specified surface temperature.

There is no transfer of mass at a solid boundary; thus, the boundary value for the species mixture  $\alpha$  is simply

$$Z_{\alpha,w} = Z_{\alpha,g} \quad (3.15)$$

## Open Boundaries

The term “open” denotes a non-solid exterior boundary of the computational domain. Gases are allowed to flow freely in and out. At these boundaries, the temperature and species mass fractions take on their respective exterior values if the flow is incoming, and take on their respective values in the grid cell adjacent to the boundary if the flow is outgoing. This is a simple upwind boundary condition.

## Specified Mass Flux

Here, the mass flux of species  $\alpha$ ,  $\dot{m}_{\alpha}''$ , is specified or computed as part of the overall solid phase calculation. To determine the mass fraction of species mixture  $\alpha$  at the boundary,  $Z_{\alpha,f}$ , the following equations must be solved iteratively

$$\sum_{\alpha} \dot{m}_{\alpha}'' = \rho_w u_n \quad (3.16)$$

$$\dot{m}_{\alpha}'' = u_n \rho_w Z_{\alpha,w} - (\rho D_{\alpha})_w \frac{Z_{\alpha,g} - Z_{\alpha,w}}{\delta n/2} \quad (3.17)$$

where  $u_n$  is the normal component of velocity at the wall pointing into the flow domain and  $\delta n/2$  is the distance between the center of the gas cell and the wall. Together with the equation of state, Eqs. (3.16) and (3.17) are solved iteratively for the unknowns  $\rho_w$ ,  $u_n$ , and  $Z_{\alpha,w}$ . The surface temperature used in the EOS depends on the thermal boundary condition.

## Mesh Interface Boundaries

In simulations involving more than one numerical mesh, information has to be passed between meshes, even when the meshes are being processed by separate computers. If two meshes abut each other, and the mesh cells are aligned and the same size, then one mesh simply uses the density and species mass fractions of the adjacent mesh as the “ghost” cell values. However, in cases where the mesh cells are not the same size, the exchange of information must be done more carefully. Consider a case where two meshes meet:

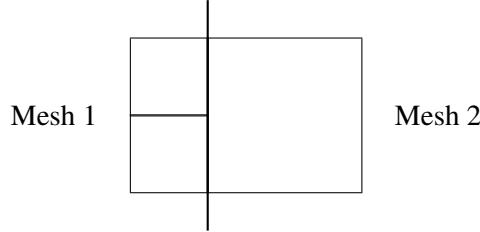


Figure 3.1: Mesh interface boundary with 2:1 refinement.

**Advective Flux Matching** We want the total and species mass fluxes between meshes to be the same. Let the density in cell  $(1, j', k')$  of Mesh 2 be denoted  $\rho_{1,j'k'}^{(2)}$ . Assume that this cell abuts two cells in Mesh 1. The densities in the two abutting cells of Mesh 1 are denoted  $\rho_{I,jk}^{(1)}$ . Note that  $j$  and  $k$  are not the same as  $j'$  and  $k'$ .  $I$  is the number of cells in the  $x$  direction of Mesh 1. The ghost cell quantities in Mesh 1 have an  $i$  index of  $I + 1$ . The ghost cell quantities in Mesh 2 have an  $i$  index of 0. We want to assert mass conservation at the mesh interface:

$$\sum_{j,k} u_{I,jk}^{(1)} \rho_{w,jk}^{(1)} \delta y^{(1)} \delta z^{(1)} = u_{0,j'k'}^{(2)} \rho_{w,j'k'}^{(2)} \delta y^{(2)} \delta z^{(2)} \quad (3.18)$$

To enforce this condition, we obtain  $\rho_{w,jk}^{(1)}$  on Mesh 1 and  $\rho_{w,j'k'}^{(2)}$  on Mesh 2 from a flux limiter (see Section 3.2.1) once data has been exchanged between meshes. If both Mesh 1 and Mesh 2 have the same grid resolution, then two layers of ghost cells are exchanged to enable second-order accuracy for advective fluxes. First-order upwinding is used at refined mesh boundaries.

**Diffusive Flux Matching** For a mesh boundary without refinement, the exchange of ghost cell information is sufficient to achieve matched diffusive fluxes computed independently by each mesh process. However, for refined meshes, special treatment is required. The strategy employed is to compute the fine mesh fluxes and average them for the coarse mesh. After the scalar transport update, the scalar values are exchanged so that the coarse mesh has the necessary fine mesh values to independently compute and average the fine mesh fluxes for matching. Referring again to Fig. 3.1, the following relationship is enforced at the mesh refinement boundary:

$$\sum_{j,k} \frac{1}{2} \left[ (\rho D_\alpha)_{1,j'k'}^{(2)} + (\rho D_\alpha)_{IBAR,jk}^{(1)} \right] \frac{Z_{\alpha,1,j'k'}^{(2)} - Z_{\alpha,IBAR,jk}^{(1)}}{\delta x_{I,jk}^{(1)}} \delta y_{jk}^{(1)} \delta z_{jk}^{(1)} = \underbrace{\left[ (\rho D_\alpha) \frac{\partial Z_\alpha}{\partial x} \right]_{0,j'k'}^{(2)}}_{\text{computed term}} \delta y_{j'k'}^{(2)} \delta z_{j'k'}^{(2)} \quad (3.19)$$

Note that the value  $Z_{\alpha,1,j'k'}^{(2)}$  is directly injected from the coarse mesh as a fine mesh ghost cell value (i.e., no interpolation is performed to the fine mesh ghost cell center; this is a first-order approximation). Similarly,

the coarse mesh diffusivity term  $(\rho D_\alpha)_{1,j'k'}^{(2)}$  is directly injected to the fine mesh ghost cell to compute the average interface value.

### Special Topic: Vertical Mesh Coarsening in Atmospheric Flows

In atmospheric flows, it may be beneficial to coarsen the mesh in the vertical direction. Matching the heat flux at the mesh interface is not trivial due to the background pressure stratification and the relative coarse grid resolution usually applied to atmospheric flow calculations. FDS does not explicitly match the flux. Instead, it relies on the temperature gradient constructed from the interior (KKG) and ghost cell (KK) values to be equivalent (additionally, the interpolation of the transport coefficient at the mesh boundary must match, but this is less of a problem than the temperature gradient). To further complicate matters, FDS always extracts the gas phase temperature value from the ideal gas law, given the cell species composition, density, and background pressure.

$$T(z) = \frac{P(z)\bar{W}(z)}{\rho(z)R} = \frac{\text{PBAR}(z)}{\text{RHO}(z)\text{RSUM}(z)} \quad (3.20)$$

At INTERPOLATED boundaries with mesh coarsening (or refinement) the positions of the ghost cell values do not match positions of the interior gas phase cell values from the neighboring mesh. This is depicted for our problem in Fig. 3.2. Let  $\delta z_{kkg}^{(1)}$  denote the cell height for the interior gas phase cell on Mesh 1, for example. To match the temperature gradient at the mesh interface we require

$$\frac{T_{kk}^{(1)} - T_{kkg}^{(1)}}{\frac{1}{2}\delta z_{kk}^{(1)} + \frac{1}{2}\delta z_{kkg}^{(1)}} = \frac{T_{kkg}^{(2)} - T_{kkg}^{(1)}}{\frac{1}{2}\delta z_{kkg}^{(2)} + \frac{1}{2}\delta z_{kkg}^{(1)}} \quad (3.21)$$

Now, suppose we are filling ghost cell values for Mesh 1. Our goal is to specify the ghost cell value of the density,  $\rho_{kk}^{(1)}$ , such that Eq. (3.21) holds. Plugging Eq. (3.20) into Eq. (3.21), and defining

$$\text{DDO} = \frac{\delta z_{kk}^{(1)} + \delta z_{kkg}^{(1)}}{\delta z_{kkg}^{(2)} + \delta z_{kkg}^{(1)}} \quad (3.22)$$

we get

$$\rho_{kk}^{(1)} = \left( \frac{P_{kk}^{(1)}}{\text{RSUM}_{kk}^{(1)}} \right) \Big/ \left\{ \left( \frac{P_{kkg}^{(1)}}{\rho_{kkg}^{(1)} \text{RSUM}_{kkg}^{(1)}} \right) + \text{DDO} \left[ \left( \frac{P_{kkg}^{(2)}}{\rho_{kkg}^{(2)} \text{RSUM}_{kkg}^{(2)}} \right) - \left( \frac{P_{kkg}^{(1)}}{\rho_{kkg}^{(1)} \text{RSUM}_{kkg}^{(1)}} \right) \right] \right\} \quad (3.23)$$

A similar expression is obtained when viewing the interpolation from Mesh 2.

Note that this interpolation may be thought of as a central difference for the temperature gradient. The corresponding density variation is nonlinear. The mass fluxes computed by the flux limiter scheme described above still match at the refined mesh interface and therefore do not require adjustment.

This special treatment of the ghost cell density may be accessed in the source code by searching on the logical flag `ATMOSPHERIC_INTERPOLATION`. By default, the procedure is invoked at vertical mesh refinement boundaries only if `STRATIFICATION=.TRUE.` and if the vertical cell spacing `DZ` on any mesh exceeds 2 m. For smaller grid spacing the spurious temperature effect is not noticeable and it is deemed more appropriate to keep the temperature variation consistent with the flux limited interface density. Setting the developer logical flag `USE_ATMOSPHERIC_INTERPOLATION` on the `WIND` line overrides the default behavior.

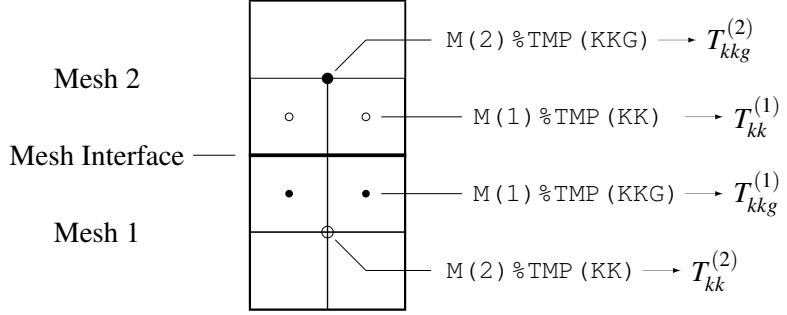


Figure 3.2: Vertical mesh coarsening. Mesh 1 is the fine mesh. Mesh 2 is the coarse mesh. In this example, the coarsening ratio is 1:2. Solid dots represent interior gas phase unknown values. Open circles represent ghost cell values.

### 3.3 The Velocity Divergence

Because of the low Mach number assumption, the velocity divergence (the rate of volumetric expansion) plays an important role in the overall solution scheme. In the FDS algorithm, the divergence is a surrogate for the energy equation. The divergence is factored out of the conservative form of the sensible enthalpy equation (see Appendix B), and when the divergence constraint is satisfied (enforced by the momentum update and solution of the Poisson equation for pressure) the conservative form of the sensible enthalpy equation is satisfied by construction.

For the  $m$ th zone, with background pressure  $\bar{p}_m$ , the divergence may be written as

$$\nabla \cdot \mathbf{u} = D - P \frac{\partial \bar{p}_m}{\partial t} \quad (3.24)$$

where

$$P = \frac{1}{\bar{p}_m} - \frac{1}{\rho c_p T} \quad (3.25)$$

and

$$\begin{aligned} D &= \frac{1}{\rho c_p T} [\dot{q}''' + \dot{q}_b''' - \nabla \cdot \dot{\mathbf{q}}'' - \mathbf{u} \cdot \nabla(\rho h_s) + w \rho_0 g_z] \\ &\quad + \frac{1}{\rho} \sum_{\alpha} \left( \frac{\bar{W}}{W_{\alpha}} - \frac{h_{s,\alpha}}{c_p T} \right) \left[ \nabla \cdot (\rho D_{\alpha} \nabla Y_{\alpha}) - \mathbf{u} \cdot \nabla(\rho Y_{\alpha}) + \dot{m}_{\alpha}''' \right] \\ &\quad + \frac{1}{\rho} \sum_{\alpha} \left( \frac{\bar{W}}{W_{\alpha}} - \frac{\int_{T_b}^T c_{p,\alpha}(T') dT'}{c_p T} \right) \dot{m}_{\alpha}''' \end{aligned} \quad (3.26)$$

#### 3.3.1 Mass and Energy Source Terms

The volumetric source terms in the divergence expression require extended discussion. The heat release rate per unit volume,  $\dot{q}'''$ , and the mass generation rate of species  $\alpha$  per unit volume,  $\dot{m}_{\alpha}'''$ , are detailed in Chapter 5, Combustion. The flux term,  $\dot{\mathbf{q}}''$ , is defined in Eq. (2.9). The radiative source term,  $\dot{q}_r'''$ , that is included in  $\nabla \cdot \dot{\mathbf{q}}''$  is discussed in Chapter 6, Thermal Radiation. The bulk heat source from Lagrangian

particles,  $\dot{q}_b''$ , which accounts for convective heat transfer and radiative absorption, is discussed in Chapter 9, Lagrangian Particles. The bulk mass source from Lagrangian particles,  $\dot{m}_{b,\alpha}'''$ , is also found in Chapter 9.

The source terms are computed in the corrector stage of the time step, following the update of the density and species mass fractions. The terms in Eq. (3.26) involving  $\dot{q}_b''$ ,  $\dot{m}_\alpha'''$ , and  $\dot{m}_{b,\alpha}'''$  are stored in an array called D\_SOURCE and applied in the construction of the divergence expression required for the corrected update of the velocity.

### 3.3.2 Diffusion Terms

The thermal and material diffusion terms of Eq. (3.26) are pure second-order central differences. For example, the thermal conduction term is differenced as follows:

$$(\nabla \cdot k \nabla T)_{ijk} = \frac{1}{\delta x} \left[ k_{i+\frac{1}{2},jk} \frac{T_{i+1,jk} - T_{ijk}}{\delta x} - k_{i-\frac{1}{2},jk} \frac{T_{ijk} - T_{i-1,jk}}{\delta x} \right] + \frac{1}{\delta y} \left[ k_{i,j+\frac{1}{2},k} \frac{T_{i,j+1,k} - T_{ijk}}{\delta y} - k_{i,j-\frac{1}{2},k} \frac{T_{ijk} - T_{i,j-1,k}}{\delta y} \right] + \frac{1}{\delta z} \left[ k_{ij,k+\frac{1}{2}} \frac{T_{ij,k+1} - T_{ijk}}{\delta z} - k_{ij,k-\frac{1}{2}} \frac{T_{ijk} - T_{ij,k-1}}{\delta z} \right] \quad (3.27)$$

The thermal conductivity at the cell interface, denoted by the  $\frac{1}{2}$  cell index, is the average of its values in the two adjacent cells.

### 3.3.3 Corrections for Numerical Mixing

The differencing of the convection terms,  $\mathbf{u} \cdot \nabla(\rho h_s)$  and  $\mathbf{u} \cdot \nabla(\rho Y_\alpha)$ , is complex. If not handled carefully, subtle issues related to numerical diffusion in the scalar transport schemes can cause significant conservation errors in the implied energy equation. The proper discretization of these terms is discussed in Appendix B.

### 3.3.4 Computing the Temperature

The mean cell gas temperature,  $T$ , is derived from the density and species mass fractions via the equation of state:

$$T_{ijk} = \frac{\bar{P}_m}{\rho_{ijk} R \sum_{\alpha=0}^{N_s} (Z_{\alpha,ijk} / W_\alpha)} \quad (3.28)$$

### 3.3.5 Sensible Enthalpy

The sensible enthalpy of the gas is a mass-weighted average of the enthalpies of the lumped species (denoted by  $\alpha$ ), which are in turn a mass-weighted average of the enthalpies of the individual gas species (denoted by  $n$ ):

$$h_s = \sum_{\alpha} Z_{\alpha} h_{s,\alpha} \quad ; \quad h_{s,\alpha} = \sum_n Y_n h_{s,n} \quad ; \quad h_{s,n}(T) = \int_{T_0}^T c_{p,n}(T') dT' \quad (3.29)$$

The values of  $h_{s,n}$  and  $c_{p,n}$  for the individual gas species are obtained by table lookup from NASA polynomials [25] or the NIST-JANAF tables [26] or The values are taken to the nearest degree Kelvin.

### 3.3.6 Computing the Background Pressure Rise

To describe how the background pressure of the  $m$ th pressure zone,  $\bar{p}_m$ , is updated in time, consider the expression for the divergence written in compact notation:

$$\nabla \cdot \mathbf{u} = D - P \frac{\partial \bar{p}_m}{\partial t} \quad (3.30)$$

The terms  $D$  and  $P$  are defined by Eqs. (3.26) and (3.25), respectively. The subscript  $m$  refers to the number of the *pressure zone*; that is, a volume within the computational domain that is allowed to have its own background pressure rise. A closed room within a building, for example, is a pressure zone. The time derivative of the background pressure of the  $m$ th pressure zone is found by integrating Eq. (3.30) over the zone volume (denoted by  $\Omega_m$ ):

$$\frac{\partial \bar{p}_m}{\partial t} = \left( \int_{\Omega_m} D \, dV - \int_{\partial \Omega_m} \mathbf{u} \cdot d\mathbf{S} \right) / \int_{\Omega_m} P \, dV \quad (3.31)$$

Equation (3.31) is essentially a consistency condition, ensuring that supplying or exhausting air or starting a fire within a sealed compartment leads to an appropriate change in the divergence within the volume.

### 3.3.7 Combining Pressure Zones

In the event that a barrier separating two pressure zones should rupture (e.g., opening a door), Eq. (3.31) is modified so that the pressure in the newly connected zones is driven towards an equilibrium pressure:

$$\bar{p}_{\text{eq}} = \sum_m \left( \bar{p}_m \int_{\Omega_m} P \, dV \right) / \sum_m \int_{\Omega_m} P \, dV \approx \frac{\sum_m V_m}{\sum_m (V_m / \bar{p}_m)} \quad (3.32)$$

Note that

$$\int_{\Omega_m} P \, dV \approx \frac{V_m}{\gamma \bar{p}_m} \quad (3.33)$$

where  $V_m$  is the volume of zone  $m$  and  $\gamma$  is the ratio of specific heats. To drive the pressure within the connected zones towards each other, a volume flow,  $\dot{V}_m^*$ , is applied to each zone. This flow is intended to move gas from zones with the higher pressures towards zones with lower pressures. Eq. (3.31) now becomes:

$$\frac{\partial \bar{p}_{\text{eq}}}{\partial t} - \frac{\bar{p}_m - \bar{p}_{\text{eq}}}{\tau} = \left( \int_{\Omega_m} D \, dV - \int_{\partial \Omega_m} \mathbf{u} \cdot d\mathbf{S} - \dot{V}_m^* \right) / \int_{\Omega_m} P \, dV \quad (3.34)$$

This equation is solved for  $\dot{V}_m^*$ . The first term on the left is the change in the equilibrium pressure with time:

$$\frac{\partial \bar{p}_{\text{eq}}}{\partial t} = \left( \sum_m \int_{\Omega_m} D \, dV - \sum_m \int_{\partial \Omega_m} \mathbf{u} \cdot d\mathbf{S} \right) / \sum_m \int_{\Omega_m} P \, dV \quad (3.35)$$

The summation is over all connected zones, and it is essentially the net change in pressure with time for the entire connected region. If there is any opening to the exterior of the computational domain, this term is set to zero and all connected zone pressures are driven towards ambient. The second term on the left forces the pressure in the  $m$ th pressure zone towards the equilibrium. The constant,  $\tau$ , is a characteristic time for the pressure to come into equilibrium. Its default value is on the order of 1 s. In reality, room pressures typically come into equilibrium very rapidly, but air movements associated with rapid changes in pressure can cause numerical instabilities.

**Note:** Because of the low Mach number assumption, FDS should not be used for rapid discharge of pressure vessels.

# Chapter 4

# Momentum Transport and Pressure

This chapter describes the solution of the momentum equation. This consists of three major parts: the LES formulation, the discretization of the flux terms, and the solution of an elliptic partial differential equation for the pressure.

## 4.1 Large Eddy Simulation (LES)

In this section, we temporarily return to formal LES filter notation and adopt Cartesian tensor index notation (repeated suffixes imply summation) in order to precisely define modeled terms. The LES equations are derived by applying a low-pass filter of width  $\Delta$  to the DNS equations. The kernel usually associated with finite volume LES is a box filter—grid resolved quantities are physically interpreted as cell means. This interpretation is somewhat misleading (see [27]), but a thorough discussion of filtering is beyond our scope, so the cell mean interpretation will suffice. In FDS, the filter width is taken to be the cube root of the cell volume,  $\Delta = V_c^{1/3}$ ,  $V_c = \delta x \delta y \delta z$ . Then for any continuous field,  $\phi$ , a filtered field is defined as

$$\bar{\phi}(x, y, z, t) \equiv \frac{1}{V_c} \int_{x-\delta x/2}^{x+\delta x/2} \int_{y-\delta y/2}^{y+\delta y/2} \int_{z-\delta z/2}^{z+\delta z/2} \phi(x', y', z', t) dx' dy' dz' \quad (4.1)$$

It is also conventional to define a mass-weighted or Favre filter such that  $\bar{\rho} \tilde{\phi} \equiv \bar{\rho} \bar{\phi}$ .

### 4.1.1 The DNS Momentum Equation

In conservative form, the DNS momentum equation for the  $i$ th component of velocity is

$$\frac{\partial \rho u_i}{\partial t} + \frac{\partial}{\partial x_j} (\rho u_i u_j) = -\frac{\partial p}{\partial x_i} - \frac{\partial \tau_{ij}}{\partial x_j} + \rho g_i + f_{d,i} + \dot{m}_b''' u_{b,i} \quad (4.2)$$

In our two-phase formulation,  $f_{d,i}$  represents the drag force due to unresolved Lagrangian particles. The bulk source term,  $\dot{m}_b''' u_{b,i}$ , accounts for the effects of evaporation or pyrolysis. For Eq. (4.2) to be applicable, the grid resolution should be smaller than the Kolmogorov scale,  $\eta$ , the length scale of the smallest turbulent eddies [15],

$$\eta \equiv (v^3 / \varepsilon)^{1/4} \quad (4.3)$$

Here,  $v$  is the kinematic viscosity and  $\varepsilon$  is the rate of viscous dissipation (the conversion of kinetic energy to heat by viscosity),

$$\varepsilon \equiv \tau_{ij} \frac{\partial u_i}{\partial x_j} = 2\mu \left( S_{ij} S_{ij} - \frac{1}{3} (\nabla \cdot \mathbf{u})^2 \right) \quad ; \quad S_{ij} \equiv \frac{1}{2} \left( \frac{\partial u_i}{\partial x_j} + \frac{\partial u_j}{\partial x_i} \right) \quad (4.4)$$

In fire scenarios,  $\eta$  is usually on the order of one millimeter. DNS is therefore impractical for all but special research flame calculations.

### 4.1.2 The LES Momentum Equation

For domain sizes ranging from meters to kilometers, the affordable grid resolution for most LES fire calculations ranges from centimeters to meters. The goal of the LES is to evolve the cell mean values of mass, momentum, and energy explicitly, while accounting for the effects that subgrid transport and chemistry have on the mean fields. To this end, we apply the box filter to the DNS equations to obtain the filtered equations. As an example, consider the momentum equation. Applying Eq. (4.1) to Eq. (4.2) results in

$$\frac{\partial \bar{\rho} \bar{u}_i}{\partial t} + \frac{\partial}{\partial x_j} (\bar{\rho} \bar{u}_i \bar{u}_j) = - \frac{\partial \bar{p}}{\partial x_i} - \frac{\partial \bar{\tau}_{ij}}{\partial x_j} + \bar{\rho} g_i + \bar{f}_{d,i} + \bar{m}_b''' \bar{u}_{b,i} \quad (4.5)$$

The cell mean value,  $\bar{\rho} \bar{u}_i \bar{u}_j$ , is not itself a primitive variable in the calculation—we have no way of computing the term under the bar to advance Eq. (4.5) in time. We must, therefore, decompose the terms, and this leads to closure problems.

The next step is simply to apply the Favre filter,

$$\frac{\partial \bar{\rho} \tilde{u}_i}{\partial t} + \frac{\partial}{\partial x_j} (\bar{\rho} \tilde{u}_i \tilde{u}_j) = - \frac{\partial \bar{p}}{\partial x_i} - \frac{\partial \bar{\tau}_{ij}}{\partial x_j} + \bar{\rho} g_i + \bar{f}_{d,i} + \bar{m}_b''' \tilde{u}_{b,i} \quad (4.6)$$

The first term is now separable, provided we have a solution for  $\bar{\rho}$ . But we still have no way to compute the correlation  $\tilde{u}_i \tilde{u}_j$  on the grid. We cannot simply use  $\tilde{u}_i \tilde{u}_j$  as a substitute (this is the old problem of “the mean of the square does not equal the square of the mean”). Instead, we define the subgrid-scale (SGS) stress:

$$\tau_{ij}^{sgs} \equiv \bar{\rho} (\tilde{u}_i \tilde{u}_j - \bar{u}_i \bar{u}_j) \quad (4.7)$$

Substituting Eq. (4.7) into Eq. (4.6) yields

$$\frac{\partial \bar{\rho} \tilde{u}_i}{\partial t} + \frac{\partial}{\partial x_j} (\bar{\rho} \tilde{u}_i \tilde{u}_j) = - \frac{\partial \bar{p}}{\partial x_i} - \frac{\partial \bar{\tau}_{ij}}{\partial x_j} - \frac{\partial \tau_{ij}^{sgs}}{\partial x_j} + \bar{\rho} g_i + \bar{f}_{d,i} + \bar{m}_b''' \tilde{u}_{b,i} \quad (4.8)$$

Equation (4.8) is what is typically referred to as the LES momentum equation (analogous to the Cauchy equation—constitutive models have not been applied). All variables are primitive or computable once we find a suitable closure for the subgrid scale stress,  $\tau_{ij}^{sgs}$ .

### Constitutive Relationship

There are a few more modifications we need to make in order to get Eq. (4.8) into shape for FDS. The first is to decompose the SGS stress and apply Newton’s law of viscosity as the constitutive relationship for the deviatoric part. Note that  $\bar{\tau}_{ij}$  is already the deviatoric part of the viscous stress. We model the total deviatoric stress as

$$\tau_{ij}^{\text{dev}} \equiv \bar{\tau}_{ij} + \tau_{ij}^{sgs} - \frac{1}{3} \tau_{kk}^{sgs} \delta_{ij} = -2(\mu + \mu_t) \left( \tilde{S}_{ij} - \frac{1}{3} (\nabla \cdot \tilde{\mathbf{u}}) \delta_{ij} \right) \quad (4.9)$$

Note that  $\delta_{ij}$  is the Kronecker delta ( $\delta_{ij} = 1$  if  $i = j$ ,  $\delta_{ij} = 0$  if  $i \neq j$ ). The turbulent viscosity,  $\mu_t$ , requires modeling, as discussed below.

## Modified Pressure Term

In LES of low-Mach flows, the isotropic part of the SGS stress must be absorbed by the pressure term. Define the subgrid kinetic energy as half the trace of the SGS stress,

$$k_{\text{sgs}} \equiv \frac{1}{2} \tau_{kk}^{\text{sgs}} \quad (4.10)$$

and define the modified filtered pressure [15] as

$$\bar{p} \equiv \bar{p} + \frac{2}{3} k_{\text{sgs}} \quad (4.11)$$

Upon substitution of Eqs. (4.9) and (4.11) into Eq. (4.8), we have

$$\frac{\partial \bar{p} \tilde{u}_i}{\partial t} + \frac{\partial}{\partial x_j} (\bar{p} \tilde{u}_i \tilde{u}_j) = -\frac{\partial \bar{p}}{\partial x_i} - \frac{\partial \tau_{ij}^{\text{dev}}}{\partial x_j} + \bar{p} g_i + \bar{f}_{d,i} + \overline{m_b''' \tilde{u}_{b,i}} \quad (4.12)$$

Notice that Eq. (4.12) closely resembles the DNS momentum equation, Eq. (4.2). For this reason, we may relax the filter formalism as we discuss the numerical details of the algorithm. The user should simply understand that in the LES context when we write  $\tau_{ij}$  we mean precisely  $\tau_{ij}^{\text{dev}}$ , and similarly for pressure in LES  $p$  refers to  $\bar{p}$ .

## Bulk Mass Source Term

When writing the momentum equation in non-conservative form, which we will do below, we must account for the introduction of mass from subgrid particles (evaporation of water droplets, for example). Using the continuity equation, Eq. (2.6), we can rewrite Eq. (4.12) as follows:

$$\bar{p} \frac{D \tilde{u}_i}{Dt} = -\frac{\partial \bar{p}}{\partial x_i} - \frac{\partial \tau_{ij}^{\text{dev}}}{\partial x_j} + \bar{p} g_i + \underbrace{\bar{f}_{d,i} + \overline{m_b''' (\tilde{u}_{b,i} - \tilde{u}_i)}}_{\bar{f}_{b,i}} \quad (4.13)$$

The last term in Eq. (4.13) is absorbed into the bulk subgrid force term,  $\bar{f}_{b,i}$ , which also accounts for drag, as discussed in Chapter 9 on Lagrangian Particles.

### 4.1.3 Production of Subgrid Kinetic Energy

The transport equation for the resolved kinetic energy per unit mass,  $K \equiv \frac{1}{2} \tilde{u}_i \tilde{u}_i$ , is derived by dotting the LES momentum equation with the resolved velocity vector. The result is

$$\begin{aligned} \bar{p} \frac{DK}{Dt} &= -\tilde{u}_i \frac{\partial \bar{p}}{\partial x_i} - \tilde{u}_i \frac{\partial \tau_{ij}^{\text{dev}}}{\partial x_j} + (\bar{p} g_i + \bar{f}_{b,i}) \tilde{u}_i \\ \bar{p} \frac{DK}{Dt} + \frac{\partial}{\partial x_j} ([\bar{p} \delta_{ij} + \tau_{ij}^{\text{dev}}] \tilde{u}_i) &= \bar{p} \frac{\partial \tilde{u}_i}{\partial x_i} + \tau_{ij}^{\text{dev}} \frac{\partial \tilde{u}_i}{\partial x_j} + (\bar{p} g_i + \bar{f}_{b,i}) \tilde{u}_i \end{aligned} \quad (4.14)$$

The terms on the left hand side represent transport. The terms on the right hand side are sources or sinks of kinetic energy. Of particular interest in LES is the *production of subgrid kinetic energy*, buried in the second RHS term. The effect of this term is to transfer energy between the resolved and unresolved scales of motion. In the classical picture of the “energy cascade”, the net transfer of energy is from large to small scales, where ultimately the motions are dissipated as heat by viscosity. In LES, however, this term may also be a source of energy, a phenomenon called *backscatter*. An issue that makes designing subgrid closures

for LES challenging is that, far from being the exception, backscatter is ubiquitous and often critical to the formation of large-scale motions (think of subgrid buoyancy-generated turbulence, e.g., the Rayleigh-Taylor instability) [28]. Most simple LES subgrid closures take production of subgrid kinetic energy to be equal to the dissipation of total kinetic energy. Using gradient diffusion for SGS closure, this assumption implies the following:

$$\begin{aligned}\tau_{ij}^{\text{dev}} \frac{\partial \tilde{u}_i}{\partial x_j} &= -2(\mu + \mu_t) \left( \tilde{S}_{ij} - \frac{1}{3} (\nabla \cdot \tilde{\mathbf{u}}) \delta_{ij} \right) \frac{\partial \tilde{u}_i}{\partial x_j} \\ &= -2(\mu + \mu_t) \left( \tilde{S}_{ij} - \frac{1}{3} (\nabla \cdot \tilde{\mathbf{u}}) \delta_{ij} \right) \tilde{S}_{ij} \\ &= -2(\mu + \mu_t) \left( \tilde{S}_{ij} \tilde{S}_{ij} - \frac{1}{3} (\nabla \cdot \tilde{\mathbf{u}})^2 \right) = -2\mu \left( S_{ij} S_{ij} - \frac{1}{3} (\nabla \cdot \mathbf{u})^2 \right) \equiv \varepsilon\end{aligned}\quad (4.15)$$

Using a model kinetic energy spectrum (see [15]), Eq. (4.15) can be used to derive theoretical values for model constants, such as the Smagorinsky constant, discussed below. See Appendix H.

## 4.2 Models for the Turbulent Viscosity

In LES, the “turbulence model” refers to the closure for SGS flux terms. In FDS, *gradient diffusion* is the turbulence model used to close both the SGS momentum and scalar flux terms. We then require a model for the turbulent transport coefficient: the turbulent (or eddy) viscosity or the turbulent (or eddy) diffusivity. The turbulent diffusivity is obtained using a constant turbulent Schmidt number (for mass diffusivity) or Prandtl number (for thermal diffusivity), as discussed below, and so the most important transport coefficient is the turbulent viscosity,  $\mu_t$ . There are several different options available that are described in this section. The Deardorff model, Sec. 4.2.3, is the default. Its selection as the default was based on comparisons with a wide variety of full-scale experiments.

### 4.2.1 Constant Coefficient Smagorinsky Model

Following the analysis of Smagorinsky [29], the eddy viscosity can be modeled as follows:

$$\mu_t = \rho (C_s \Delta)^2 |S| \quad ; \quad |S| = \left( 2S_{ij}S_{ij} - \frac{2}{3}(\nabla \cdot \mathbf{u})^2 \right)^{\frac{1}{2}} \quad (4.16)$$

where  $C_s = 0.2$  is a constant and  $\Delta = (\delta x \delta y \delta z)^{1/3}$  is the filter width. This model was used in FDS versions 1 through 5. The value of  $C_s = 0.2$  is nominally the value obtained from Lilly’s analysis [30] (production equals dissipation) for a spectral cutoff filter (implicit filters for energy conserving schemes more closely resemble a spectral cutoff than a box filter [27]). The constant value is derived theoretically in Appendix H and also confirmed in tests of decaying isotropic turbulence in the FDS Verification Guide [31].

### 4.2.2 Dynamic Smagorinsky Model

For the dynamic Smagorinsky model [32, 33], the coefficient  $C_s$  in Eq. (4.16) is no longer taken as a constant, but rather computed based on local flow conditions.

### 4.2.3 Deardorff’s Model (Default)

By default, FDS uses a variation of Deardorff’s model [14]:

$$\mu_t = \rho C_v \Delta \sqrt{k_{sgs}} \quad ; \quad k_{sgs} = \frac{1}{2} ((\bar{u} - \hat{u})^2 + (\bar{v} - \hat{v})^2 + (\bar{w} - \hat{w})^2) \quad (4.17)$$

where  $\bar{u}$  is the average value of  $u$  at the grid cell center (representing the LES filtered velocity at length scale  $\Delta$ ) and  $\hat{u}$  is a weighted average of  $u$  over the adjacent cells (representing a test-filtered field at length scale  $2\Delta$ ):

$$\bar{u}_{ijk} = \frac{u_{ijk} + u_{i-1,jk}}{2} \quad ; \quad \hat{u}_{ijk} = \frac{\bar{u}_{ijk}}{2} + \frac{\bar{u}_{i-1,jk} + \bar{u}_{i+1,jk}}{4} \quad (4.18)$$

The terms  $\hat{v}$  and  $\hat{w}$  are defined similarly. The model constant is set to the literature value  $C_v = 0.1$  [15]. The constant value is also derived theoretically in Appendix H. The algebraic form of subgrid kinetic energy is based on the ideas presented in the scale-similarity model of Bardina et al. [34]. (Note that Deardorff [14] solved a transport equation for  $k_{sgs}$ .)

#### 4.2.4 Vreman's Model

Vreman's eddy viscosity model [35] is given by

$$\mu_t = \rho c \sqrt{\frac{B_\beta}{\alpha_{ij} \alpha_{ij}}} \quad (4.19)$$

where

$$B_\beta = \beta_{11}\beta_{22} - \beta_{12}^2 + \beta_{11}\beta_{33} - \beta_{13}^2 + \beta_{22}\beta_{33} - \beta_{23}^2 \quad ; \quad \beta_{ij} = \Delta_m^2 \alpha_{mi} \alpha_{mj} \quad (4.20)$$

$$\alpha_{ij} = \frac{\partial u_j}{\partial x_i} \quad (4.21)$$

The notation is selected to exactly match Vreman's paper [35]. The model considers a possible anisotropy in the filter width in direction  $m$ ; contraction on  $m$  is implied in Eq. (4.20). The basic idea behind Vreman's model is to expand the velocity field in a Taylor series and to test filter this field analytically, thus avoiding the expensive explicit test filtering operations necessary in the dynamic model. Therefore, this model is inexpensive. Unlike constant coefficient Smagorinsky, however, Vreman's model is convergent, making it applicable to highly resolved LES calculations.

The model constant may be related to the Smagorinsky constant,  $c \approx 2.5 C_s^2$ . Since Vreman's model is most applicable to high resolution cases, we base the coefficient off of  $C_s = 0.17$ , which yields accurate results for highly resolved decaying isotropic turbulence (see the FDS Verification Guide [31]). The default Vreman constant is therefore set to  $c = 0.07$ .

#### 4.2.5 Wall-Adapting Local Eddy-viscosity (WALE) Model

The Wall-Adapting Local Eddy-viscosity, or WALE, model of Nicoud and Ducros [36] was originally conceived as a method for properly scaling the eddy viscosity in the vicinity of a wall. While the invariant used in the Smagorinsky model  $|S|$  is  $O(1)$  near a wall, the invariant designed for WALE correctly scales as  $O(y^3)$ , where  $y$  is the distance from the wall. The WALE model is used for the eddy viscosity in the first off-wall grid cell. The dynamic turbulent viscosity is written as follows:

$$\mu_t = \rho (C_w \Delta)^2 \frac{(S_{ij}^d S_{ij}^d)^{3/2}}{(S_{ij} S_{ij})^{5/2} + (S_{ij}^d S_{ij}^d)^{5/2}} \quad (4.22)$$

where

$$S_{ij}^d S_{ij}^d = \frac{1}{6} (S^2 S^2 + \Omega^2 \Omega^2) + \frac{2}{3} S^2 \Omega^2 + 2 I V_{S\Omega} \quad (4.23)$$

$$S^2 = S_{ij} S_{ij} \quad ; \quad \Omega^2 = \Omega_{ij} \Omega_{ij} \quad ; \quad I V_{S\Omega} = S_{ik} S_{kj} \Omega_{jl} \Omega_{li}$$

The strain and rotation tensors are given by

$$S_{ij} = \frac{1}{2} \left( \frac{\partial u_i}{\partial x_j} + \frac{\partial u_j}{\partial x_i} \right) \quad ; \quad \Omega_{ij} = \frac{1}{2} \left( \frac{\partial u_i}{\partial x_j} - \frac{\partial u_j}{\partial x_i} \right) \quad (4.24)$$

Nicoud and Ducros [36] suggest the model constant should be in the range  $0.55 \leq C_w \leq 0.60$ . FDS uses  $C_w = 0.60$  based on results for decaying isotropic turbulence [31]. It is apparent that WALE may also be used as a bulk flow eddy-viscosity model. However, this has not been tested in fire applications.

## 4.2.6 Thermal Conduction and Gas Species Diffusion

The other diffusive parameters, the thermal conductivity and mass diffusivity, are related to the turbulent viscosity by

$$k_t = \frac{\mu_t c_p}{\text{Pr}_t} ; \quad (\rho D)_t = \frac{\mu_t}{\text{Sc}_t} \quad (4.25)$$

The turbulent Prandtl number  $\text{Pr}_t$  and the turbulent Schmidt number  $\text{Sc}_t$  are assumed to be constant for a given scenario. The default value is 0.5 for both. Justification for these values is given in [37] based on smoke plume simulations.

## 4.2.7 Numerical Implementation

In the discretized form of the momentum equation, the modeled viscosity is defined at cell centers. For example, the constant coefficient Smagorinsky model takes on the following form:

$$\mu_{ijk} = \rho_{ijk} (C_s \Delta)^2 |S| \quad (4.26)$$

where  $C_s$  is an empirical constant,  $\Delta = (\delta x \delta y \delta z)^{\frac{1}{3}}$ , and

$$|S|^2 = 2 \left( \frac{\partial u}{\partial x} \right)^2 + 2 \left( \frac{\partial v}{\partial y} \right)^2 + 2 \left( \frac{\partial w}{\partial z} \right)^2 + \left( \frac{\partial u}{\partial y} + \frac{\partial v}{\partial x} \right)^2 + \left( \frac{\partial u}{\partial z} + \frac{\partial w}{\partial x} \right)^2 + \left( \frac{\partial v}{\partial z} + \frac{\partial w}{\partial y} \right)^2 - \frac{2}{3} (\nabla \cdot \mathbf{u})^2 \quad (4.27)$$

The quantity  $|S|$  consists of second order spatial differences averaged at cell centers. For example

$$\frac{\partial u}{\partial x} \approx \frac{u_{ijk} - u_{i-1,jk}}{\delta x_i} \quad (4.28)$$

$$\frac{\partial u}{\partial y} \approx \frac{1}{2} \left( \frac{u_{i,j+1,k} - u_{ijk}}{\delta y_{j+\frac{1}{2}}} + \frac{u_{ijk} - u_{i,j-1,k}}{\delta y_{j-\frac{1}{2}}} \right) \quad (4.29)$$

The divergence is described in Sec. 3.3.2.

## 4.2.8 Transport Coefficients for Direct Numerical Simulation (DNS)

There are some flow scenarios where it is possible to use the molecular properties  $\mu$ ,  $k$  and  $D$  directly. Usually, this means that the numerical grid cells are on the order of 1 mm or less, and the simulation is regarded as a Direct Numerical Simulation (DNS). For a DNS, the viscosity, thermal conductivity and material diffusivity are approximated from kinetic theory because the temperature dependence of each is important in combustion scenarios. The viscosity of the species  $\alpha$  is given by

$$\mu_\alpha = \frac{26.69 \times 10^{-7} (W_\alpha T)^{\frac{1}{2}}}{\sigma_\alpha^2 \Omega_v} \quad [=] \quad \text{kg}/(\text{ms}) \quad (4.30)$$

where  $\sigma_\alpha$  is the Lennard-Jones hard-sphere diameter ( $\text{\AA}$ ) and  $\Omega_v$  is the collision integral, an empirical function of the temperature  $T$ . The thermal conductivity of species  $\alpha$  is given by

$$k_\alpha = \frac{\mu_\alpha c_{p,\alpha}}{\text{Pr}_\alpha} \quad [=] \quad \text{W}/(\text{mK}) \quad (4.31)$$

Note if  $\text{Pr}_\alpha$  is not defined, the value for nitrogen (0.71) is assumed. The viscosity and thermal conductivity of a gas mixture are given by [38]

$$\mu_{\text{DNS}} = \frac{\sum_\alpha \mu_\alpha X_\alpha M_\alpha^{1/2}}{\sum_\alpha X_\alpha M_\alpha^{1/2}} \quad ; \quad k_{\text{DNS}} = \frac{\sum_\alpha k_\alpha X_\alpha M_\alpha^{1/2}}{\sum_\alpha X_\alpha M_\alpha^{1/2}} \quad (4.32)$$

The binary diffusion coefficient of species  $\alpha$  diffusing into species  $\beta$  is given by

$$D_{\alpha\beta} = \frac{2.66 \times 10^{-7} T^{3/2}}{W_{\alpha\beta}^{1/2} \sigma_{\alpha\beta}^2 \Omega_D} \quad [=] \quad \text{m}^2/\text{s} \quad (4.33)$$

where  $W_{\alpha\beta} = 2(1/W_\alpha + 1/W_\beta)^{-1}$ ,  $\sigma_{\alpha\beta} = (\sigma_\alpha + \sigma_\beta)/2$ , and  $\Omega_D$  is the diffusion collision integral, an empirical function of the temperature,  $T$  [39]. It is assumed that nitrogen is the dominant species in any combustion scenario considered here, thus the diffusion coefficient in the species mass conservation equations is that of the given species diffusing into nitrogen

$$(\rho D)_{\alpha,\text{DNS}} = \rho D_{\alpha,0} \quad (4.34)$$

where species 0 is nitrogen.

## 4.3 Coupling the Velocity and Pressure

This section explains how the momentum equation is written in finite difference form, and how the solution of the momentum equation requires the solution of an elliptic PDE for the pressure.

### 4.3.1 Simplifications of the Momentum Equation

The momentum equation in conservative form is written:

$$\frac{\partial(\rho\mathbf{u})}{\partial t} + \nabla \cdot \rho\mathbf{u}\mathbf{u} + \nabla p = \rho\mathbf{g} + \mathbf{f}_b + \nabla \cdot \boldsymbol{\tau}_{ij} \quad (4.35)$$

First, we start with the non-conservative form of the momentum equation introduced above (see Eq. 4.13)

$$\rho \left( \frac{\partial \mathbf{u}}{\partial t} + (\mathbf{u} \cdot \nabla) \mathbf{u} \right) + \nabla p = \rho\mathbf{g} + \mathbf{f}_b + \nabla \cdot \boldsymbol{\tau}_{ij} \quad (4.36)$$

Note that all momentum exchange with Lagrangian particles is represented by the force term,  $\mathbf{f}_b$ . Next, we make the following substitutions:

1. Subtract the hydrostatic pressure gradient,  $\rho_0(z)\mathbf{g}$ , from both sides. Note that  $\nabla p = \rho_0\mathbf{g} + \nabla\tilde{p}$ , and  $\rho_0(z)$  is the density profile of the ambient atmosphere.
  2. Apply the vector identity:  $(\mathbf{u} \cdot \nabla) \mathbf{u} = \nabla|\mathbf{u}|^2/2 - \mathbf{u} \times \boldsymbol{\omega}$ .
  3. Divide all terms by the density,  $\rho$ .
  4. Decompose the pressure term:
- $$\frac{1}{\rho} \nabla \tilde{p} = \nabla \left( \frac{\tilde{p}}{\rho} \right) - \tilde{p} \nabla \left( \frac{1}{\rho} \right) \quad (4.37)$$
5. Define  $H \equiv |\mathbf{u}|^2/2 + \tilde{p}/\rho$ .

Now the momentum equation can be written

$$\underbrace{\frac{\partial \mathbf{u}}{\partial t} - \mathbf{u} \times \boldsymbol{\omega} - \frac{1}{\rho} \left[ (\rho - \rho_0)\mathbf{g} + \mathbf{f}_b + \nabla \cdot \boldsymbol{\tau}_{ij} \right]}_{\mathbf{F}_A} - \tilde{p} \nabla \left( \frac{1}{\rho} \right) + \nabla H = 0 \quad (4.38)$$

Note that the subscripts A and B for the vector  $\mathbf{F}$  denote Advective and Baroclinic. As will be seen in the following sections, it is convenient to group the various terms of the momentum equation into these two terms.

### 4.3.2 Finite-Difference Approximation of the Momentum Equation

As discussed in the previous section, it is convenient to write the momentum equation in the form:

$$\frac{\partial \mathbf{u}}{\partial t} + \mathbf{F} + \nabla H = 0 \quad ; \quad \mathbf{F} = \mathbf{F}_A + \mathbf{F}_B \quad (4.39)$$

The advective and baroclinic terms,  $\mathbf{F}_A$  and  $\mathbf{F}_B$ , are expanded as:

$$F_{A,x} = w\omega_y - v\omega_z - \frac{1}{\rho} \left( (\rho - \rho_n)g_x + f_x + \frac{\partial \tau_{xx}}{\partial x} + \frac{\partial \tau_{xy}}{\partial y} + \frac{\partial \tau_{xz}}{\partial z} \right) \quad ; \quad F_{B,x} = -\tilde{p} \frac{\partial}{\partial x} \left( \frac{1}{\rho} \right) \quad (4.40)$$

$$F_{A,y} = u \omega_z - w \omega_x - \frac{1}{\rho} \left( (\rho - \rho_n) g_y + f_y + \frac{\partial \tau_{yx}}{\partial x} + \frac{\partial \tau_{yy}}{\partial y} + \frac{\partial \tau_{yz}}{\partial z} \right) ; \quad F_{B,y} = -\tilde{p} \frac{\partial}{\partial y} \left( \frac{1}{\rho} \right) \quad (4.41)$$

$$F_{A,z} = v \omega_x - u \omega_y - \frac{1}{\rho} \left( (\rho - \rho_n) g_z + f_z + \frac{\partial \tau_{zx}}{\partial x} + \frac{\partial \tau_{zy}}{\partial y} + \frac{\partial \tau_{zz}}{\partial z} \right) ; \quad F_{B,z} = -\tilde{p} \frac{\partial}{\partial z} \left( \frac{1}{\rho} \right) \quad (4.42)$$

The term  $\nabla H$  is referred to as the pressure gradient, even though, as discussed above,  $H$  is not actually a pressure. Its discretization is:

$$\frac{\partial u}{\partial t} + F_{x,ijk} + \frac{H_{i+1,jk} - H_{ijk}}{\delta x} = 0 \quad (4.43)$$

$$\frac{\partial v}{\partial t} + F_{y,ijk} + \frac{H_{i,j+1,k} - H_{ijk}}{\delta y} = 0 \quad (4.44)$$

$$\frac{\partial w}{\partial t} + F_{z,ijk} + \frac{H_{ij,k+1} - H_{ijk}}{\delta z} = 0 \quad (4.45)$$

where  $H_{ijk}$  is taken at the center of cell  $ijk$ ,  $u_{ijk}$  and  $F_{x,ijk}$  are taken at the side of the cell facing in the forward  $x$  direction,  $v_{ijk}$  and  $F_{y,ijk}$  at the side facing in the forward  $y$  direction, and  $w_{ijk}$  and  $F_{z,ijk}$  at the side facing in the forward  $z$  (vertical) direction.

For the discretization of  $\mathbf{F}_A$ , the components of the vorticity ( $\omega_x, \omega_y, \omega_z$ ) are located at cell edges pointing in the  $x, y$  and  $z$  directions, respectively. The same is true for the off-diagonal terms of the viscous stress tensor:  $\tau_{zy} = \tau_{yz}$ ,  $\tau_{xz} = \tau_{zx}$ , and  $\tau_{xy} = \tau_{yx}$ . The diagonal components of the stress tensor,  $\tau_{xx}$ ,  $\tau_{yy}$ , and  $\tau_{zz}$ , and the external force components,  $f_x$ ,  $f_y$ , and  $f_z$ , are located at their respective cell faces.

$$\begin{aligned} F_{A,x,ijk} &= \frac{1}{2} \left( w_{i+\frac{1}{2},jk} \omega_{y,ijk} + w_{i+\frac{1}{2},j,k-1} \omega_{y,i,j-1,k} \right) - \frac{1}{2} \left( v_{i+\frac{1}{2},jk} \omega_{z,ijk} + v_{i+\frac{1}{2},j-1,k} \omega_{z,i,j-1,k} \right) \\ &\quad - \frac{1}{\rho_{i+\frac{1}{2},jk}} \left( f_{x,ijk} + \frac{\tau_{xx,i+1,jk} - \tau_{xx,ijk}}{\delta x} + \frac{\tau_{xy,ijk} - \tau_{xy,i,j-1,k}}{\delta y} + \frac{\tau_{xz,ijk} - \tau_{xz,i,j,k-1}}{\delta z} \right) \end{aligned} \quad (4.46)$$

$$\begin{aligned} F_{A,y,ijk} &= \frac{1}{2} \left( u_{i,j+\frac{1}{2},k} \omega_{z,ijk} + u_{i-1,j+\frac{1}{2},k} \omega_{z,i-1,jk} \right) - \frac{1}{2} \left( w_{i,j+\frac{1}{2},k} \omega_{x,ijk} + w_{i,j+\frac{1}{2},k-1} \omega_{x,i,j-1,k} \right) \\ &\quad - \frac{1}{\rho_{i,j+\frac{1}{2},k}} \left( f_{y,ijk} + \frac{\tau_{yx,ijk} - \tau_{yx,i-1,jk}}{\delta x} + \frac{\tau_{yy,ijk} - \tau_{yy,i,j+1,k}}{\delta y} + \frac{\tau_{yz,ijk} - \tau_{yz,i,j,k-1}}{\delta z} \right) \end{aligned} \quad (4.47)$$

$$\begin{aligned} F_{A,z,ijk} &= \frac{1}{2} \left( v_{ij,k+\frac{1}{2}} \omega_{x,ijk} + v_{i,j-1,k+\frac{1}{2}} \omega_{x,i,j-1,k} \right) - \frac{1}{2} \left( u_{ij,k+\frac{1}{2}} \omega_{y,ijk} + u_{i-1,j,k+\frac{1}{2}} \omega_{y,i-1,jk} \right) \\ &\quad - \frac{1}{\rho_{ij,k+\frac{1}{2}}} \left( f_{z,ijk} + \frac{\tau_{zx,ijk} - \tau_{zx,i-1,jk}}{\delta x} + \frac{\tau_{zy,ijk} - \tau_{zy,i,j-1,k}}{\delta y} + \frac{\tau_{zz,ijk} - \tau_{zz,i,j,k+1}}{\delta z} \right) \end{aligned} \quad (4.48)$$

The components of the vorticity vector are discretized:

$$\omega_{x,ijk} = \frac{w_{i,j+1,k} - w_{ijk}}{\delta y} - \frac{v_{ij,k+1} - v_{ijk}}{\delta z} \quad (4.49)$$

$$\omega_{y,ijk} = \frac{u_{ij,k+1} - u_{ijk}}{\delta z} - \frac{w_{i+1,jk} - w_{ijk}}{\delta x} \quad (4.50)$$

$$\omega_{z,ijk} = \frac{v_{i+1,jk} - v_{ijk}}{\delta x} - \frac{u_{i,j+1,k} - u_{ijk}}{\delta y} \quad (4.51)$$

The components of the viscous stress tensor are discretized:

$$\tau_{xx,ijk} = \mu_{ijk} \left( \frac{4}{3} (\nabla \cdot \mathbf{u})_{ijk} - 2 \frac{v_{ijk} - v_{i,j-1,k}}{\delta y} - 2 \frac{w_{ijk} - w_{ij,k-1}}{\delta z} \right) \quad (4.52)$$

$$\tau_{yy,ijk} = \mu_{ijk} \left( \frac{4}{3} (\nabla \cdot \mathbf{u})_{ijk} - 2 \frac{u_{ijk} - u_{i-1,jk}}{\delta x} - 2 \frac{w_{ijk} - w_{ij,k-1}}{\delta z} \right) \quad (4.53)$$

$$\tau_{zz,ijk} = \mu_{ijk} \left( \frac{4}{3} (\nabla \cdot \mathbf{u})_{ijk} - 2 \frac{u_{ijk} - u_{i-1,jk}}{\delta x} - 2 \frac{v_{ijk} - v_{i,j-1,k}}{\delta y} \right) \quad (4.54)$$

$$\tau_{xy,ijk} = \tau_{yx,ijk} = \mu_{i+\frac{1}{2},j+\frac{1}{2},k} \left( \frac{u_{i,j+1,k} - u_{ijk}}{\delta y} + \frac{v_{i+1,jk} - v_{ijk}}{\delta x} \right) \quad (4.55)$$

$$\tau_{xz,ijk} = \tau_{zx,ijk} = \mu_{i+\frac{1}{2},j,k+\frac{1}{2}} \left( \frac{u_{ij,k+1} - u_{ijk}}{\delta z} + \frac{w_{i+1,jk} - w_{ijk}}{\delta x} \right) \quad (4.56)$$

$$\tau_{yz,ijk} = \tau_{zy,ijk} = \mu_{i,j+\frac{1}{2},k+\frac{1}{2}} \left( \frac{v_{ij,k+1} - v_{ijk}}{\delta z} + \frac{w_{i,j+1,k} - w_{ijk}}{\delta y} \right) \quad (4.57)$$

The components of the baroclinic term,  $\mathbf{F}_B$ , are discretized:

$$F_{B,x,ijk} = - \frac{\tilde{p}_{i+1,jk} \rho_{ijk} + \tilde{p}_{ijk} \rho_{i+1,jk}}{\rho_{ijk} + \rho_{i+1,jk}} \frac{1}{\delta x} \left( \frac{1}{\rho_{i+1,jk}} - \frac{1}{\rho_{ijk}} \right) \quad (4.58)$$

$$F_{B,y,ijk} = - \frac{\tilde{p}_{i,j+1,k} \rho_{ijk} + \tilde{p}_{ijk} \rho_{i,j+1,k}}{\rho_{ijk} + \rho_{i,j+1,k}} \frac{1}{\delta y} \left( \frac{1}{\rho_{i,j+1,k}} - \frac{1}{\rho_{ijk}} \right) \quad (4.59)$$

$$F_{B,z,ijk} = - \frac{\tilde{p}_{ij,k+1} \rho_{ijk} + \tilde{p}_{ijk} \rho_{ij,k+1}}{\rho_{ijk} + \rho_{ij,k+1}} \frac{1}{\delta z} \left( \frac{1}{\rho_{ij,k+1}} - \frac{1}{\rho_{ijk}} \right) \quad (4.60)$$

Notice the somewhat unusual discretization of the term  $\tilde{p} \nabla(1/\rho)$ , which is needed to be consistent with the discretization of the other two terms in Eq. (4.37):

$$\begin{aligned} \frac{1}{(\rho_{ijk} + \rho_{i+1,jk})/2} \frac{\tilde{p}_{i+1,jk} - \tilde{p}_{ijk}}{\delta x} &= \frac{1}{\delta x} \left( \frac{\tilde{p}_{i+1,jk}}{\rho_{i+1,jk}} - \frac{\tilde{p}_{ijk}}{\rho_{ijk}} \right) \\ &\quad - \frac{\tilde{p}_{i+1,jk} \rho_{ijk} + \tilde{p}_{ijk} \rho_{i+1,jk}}{\rho_{ijk} + \rho_{i+1,jk}} \frac{1}{\delta x} \left( \frac{1}{\rho_{i+1,jk}} - \frac{1}{\rho_{ijk}} \right) \end{aligned} \quad (4.61)$$

### 4.3.3 The Poisson Equation for Pressure

Before the components of velocity can be advanced in time, an elliptic partial differential equation (known as a Poisson equation) must be solved for the pressure term,  $H$ . This equation is formed by taking the divergence of the momentum equation:

$$\nabla^2 H = - \frac{\partial(\nabla \cdot \mathbf{u})}{\partial t} - \nabla \cdot (\mathbf{F}_A + \mathbf{F}_B) \quad (4.62)$$

Note that the perturbation pressure  $\tilde{p}$  appears on both sides of Eq. (4.62). The value of  $\tilde{p}$  in  $\mathbf{F}_B$  is taken from the last computed  $H$ . The pressure on the left hand side (incorporated in the variable  $H$ ) is solved

for directly. The reason for the decomposition of the pressure term is so that the linear algebraic system arising from the discretization of Eq. (4.62) has constant coefficients (i.e., it is *separable*) and can be solved to machine accuracy by a fast, direct (i.e., non-iterative) method that utilizes Fast Fourier Transforms (FFT). As will be discussed below, the Poisson equation is solved multiple times, each time driving the old and new values of  $\tilde{p}$  closer together.

The discretized form of the Poisson equation is

$$\begin{aligned} & \frac{H_{i+1,jk} - 2H_{ijk} + H_{i-1,jk}}{\delta x^2} + \frac{H_{i,j+1,k} - 2H_{ijk} + H_{i,j-1,k}}{\delta y^2} + \frac{H_{ij,k+1} - 2H_{ijk} + H_{ij,k-1}}{\delta z^2} \\ &= -\frac{F_{x,ijk} - F_{x,i-1,jk}}{\delta x} - \frac{F_{y,ijk} - F_{y,i,j-1,k}}{\delta y} - \frac{F_{z,ijk} - F_{z,i,j,k-1}}{\delta z} - \frac{\delta}{\delta t}(\nabla \cdot \mathbf{u})_{ijk} \end{aligned} \quad (4.63)$$

This elliptic partial differential equation is solved using a direct FFT-based solver [16] that is part of a library of routines for solving elliptic PDEs called CRAYFISHPAK<sup>1</sup>. To ensure that the divergence of the fluid is consistent with the definition given in Eq. (3.24), the time derivative of the divergence is defined

$$\frac{\delta}{\delta t}(\nabla \cdot \mathbf{u})_{ijk} \equiv \frac{(\nabla \cdot \mathbf{u})_{ijk}^* - (\nabla \cdot \mathbf{u})_{ijk}^n}{\delta t} \quad (4.64)$$

at the predictor step, and then

$$\frac{\delta}{\delta t}(\nabla \cdot \mathbf{u})_{ijk} \equiv \frac{(\nabla \cdot \mathbf{u})_{ijk}^{n+1} - \frac{1}{2}[(\nabla \cdot \mathbf{u})_{ijk}^* + (\nabla \cdot \mathbf{u})_{ijk}^n]}{\delta t/2} \quad (4.65)$$

at the corrector step. By construction, the thermodynamic divergence defined in Eq. (3.24) is identically equal to the divergence defined by

$$(\nabla \cdot \mathbf{u})_{ijk} = \frac{u_{ijk} - u_{i-1,jk}}{\delta x} + \frac{v_{ijk} - v_{i,j-1,k}}{\delta y} + \frac{w_{ijk} - w_{i,j,k-1}}{\delta z} \quad (4.66)$$

The equivalence of the two definitions of the divergence is a result of the form of the discretized equations, the time-stepping scheme, and the direct solution of the Poisson equation for the pressure.

The following sections describe how the boundary conditions for the pressure equation are specified.

## Open Boundary Conditions (General)

An open boundary is where fluid is allowed to flow into or out of the computational domain depending on the local pressure gradient. The boundary condition for the pressure depends on whether the local flow is incoming or outgoing. In either case, it is assumed that the quantity,  $H = \tilde{p}/\rho + |\mathbf{u}|^2/2$ , remains constant along a streamline. It is also assumed that the pressure perturbation at the boundary is a user-specified input,  $\tilde{p}_{\text{ext}}$ , that is zero by default. The Poisson solver for  $H$  requires a Dirichlet condition at an open boundary; that is, its value is specified at the external boundary of the mesh. As an example, consider the boundary,  $x = x_{\min}$ . The boundary value of  $H$  is given by the following expressions depending on the direction of the flow across the external cell face:

$$\text{BXS}(J,K) \equiv H_{\frac{1}{2},jk} = \begin{cases} \frac{\tilde{p}_{\text{ext}}}{\rho_{1,jk}} + \frac{1}{2}(\bar{u}_{1,jk}^2 + \bar{v}_{1,jk}^2 + \bar{w}_{1,jk}^2) & \text{outgoing} \\ \frac{\tilde{p}_{\text{ext}}}{\rho_{\infty}} + \frac{1}{2}(u_{\infty}^2 + v_{\infty}^2 + w_{\infty}^2) & \text{incoming} \end{cases} \quad (4.67)$$

---

<sup>1</sup>CRAYFISHPAK, a vectorized form of the elliptic equation solver FISHPAK, was originally developed at the National Center for Atmospheric Research (NCAR) in Boulder, Colorado.

`BXS` is the name of the array sent to the pressure solver. The bar over the velocity components indicates an average over the respective faces of the grid cell adjacent to the boundary. The subscript  $\infty$  denotes user-specified far field velocity and density values. Typically, the far field velocity is zero, but for simulations involving an external wind, these values can be specified accordingly.

### Open Boundary Conditions (Wind)

If the user specifies a wind `SPEED`, a slightly different `OPEN` boundary condition is invoked. The outflow boundary condition is the same as that shown in Eq. (4.67). At the inflow boundary, the mean viscous and convective forces are assumed small, and a simplified momentum equation is used to develop a boundary value for  $H$ , namely,  $\partial u_i / \partial t = -\partial H / \partial x_i$ . For example, the value of  $H$  at the lower  $x$  boundary,  $H_{\frac{1}{2},jk}$ , is set so as to approximate a Neumann boundary with a specified velocity component normal to the boundary:

$$\text{BXS}(J,K) \equiv H_{\frac{1}{2},jk} = H_{1,jk}^n + \frac{\delta x}{2} \left[ \frac{u_\infty(z,t) - u_{0,jk}^n}{\delta t} \right] \quad (4.68)$$

where  $u_\infty(z,t)$  is the prescribed external wind field velocity component at height  $z$  and time  $t$ . Note that the tangential components of the velocity vector specified by the wind field are also applied on external domain boundaries as discussed in Sec. 4.4.5.

### Solid Boundary Conditions

Boundary conditions at a solid surface fall into three distinct categories:

1. External boundary where the entire face of the mesh is either solid or a forced flow.
2. External boundary where there is a mix of open and solid surfaces.
3. Internal solid obstructions.

**Case 1:** FDS uses a direct Poisson solver that requires that one specify either Neumann (specified normal gradient) or Dirichlet (specified value) boundary conditions at the exterior of the mesh. If an entire face of the mesh is a solid or forced flow boundary, we can use the Neumann boundary condition for the entire face. For example, at the  $x = x_{\max}$  boundary we can specify the normal gradient of  $H$ :

$$\text{BXF}(J,K) \equiv \frac{H_{I+1,jk}^n - H_{I,jk}^n}{\delta x} = -F_{x,I,jk}^n - \frac{u_{I,jk}^* - u_{I,jk}^n}{\delta t} \quad (4.69)$$

where `BXF` is the name of the boundary condition array sent to the pressure solver,  $H_{I+1,jk}$  lives in the center of the ghost cell to the right of the boundary,  $H_{I,jk}$  lives in the center of cell to the left of the boundary,  $F_{x,I,jk}^n$  is the  $x$ -component of  $\mathbf{F}$  at the vent or solid wall at the start of the time step, and  $u_{I,jk}^*$  is the user-specified value of the  $x$ -component of velocity at the next time step.

Note that for Neumann pressure boundary, there are several options for computing the boundary force term (it could even be set to zero). What matters is, whatever value is used for the force here, the same value must be used in the velocity predictor and corrector steps, Eqs. (4.88) and (4.89). We choose to compute the boundary force term using the previous value of the normal pressure gradient, just as we would for an immersed boundary surface:

$$F_{x,I,jk}^{n,k} = -\frac{H_{I+1,jk}^{n,k-1} - H_{I,jk}^{n,k-1}}{\delta x} - \frac{u_{I,jk}^* - u_{I,jk}^n}{\delta t} \quad (4.70)$$

For the corrector step:

$$\text{BXF}(J, K) \equiv \frac{H_{I+1,jk}^* - H_{I,jk}^*}{\delta x} = -F_{x,I,jk}^* - \frac{u_{I,jk}^{n+1} - \frac{1}{2}(u_{I,jk}^* + u_{I,jk}^n)}{\delta t/2} \quad (4.71)$$

The normal velocity component at the next time step,  $u_{I,jk}^{n+1}$ , is exactly (to machine accuracy) the specified value. If the boundary is a solid wall, then this value starts and remains zero. If the boundary is a forced flow vent, then this value follows the user-specified time history.

**Case 2:** At exterior mesh faces with a mix of solid and open boundaries, we must apply Dirichlet boundary conditions at all cells, meaning that  $H$  is specified rather than its gradient. Consider the same mesh face as in Case 1. As with the previous case, we first modify the flux term using a previously computed value of the pressure and the desired time derivative of the velocity component:

$$F_{x,I,jk}^{n,k} = -\frac{H_{I+1,jk}^{n,k-1} - H_{I,jk}^{n,k-1}}{\delta x} - \frac{u_{I,jk}^* - u_{I,jk}^n}{\delta t} \quad (4.72)$$

Next, the value of  $H$  is specified at the mesh boundary:

$$\text{BXF}(J, K) \equiv H_{I+\frac{1}{2},jk}^{n,k} = \frac{H_{I,jk}^{n,k-1} + H_{I+1,jk}^{n,k-1}}{2} + \frac{\delta x}{4\delta t} (u_{I,jk}^{*,k-1} - u_{I,jk}^*) \quad (4.73)$$

The superscript  $k$  is an iterative index. We use the interface value of  $H$  from the previous iteration to estimate its value at the current. The term,  $u_{I,jk}^{*,k-1}$ , is a first-order estimate of the desired normal velocity component at the next time step,  $u_I^*$ . The purpose of the second term on the right hand side of Eq. (4.73) is demonstrated by summing Eq. (4.72) and Eq. (4.73) which leads to:

$$u_{I,jk}^* = u_{I,jk}^n + \frac{u_{I,jk}^{*,k-1} - u_{I,jk}^*}{2} - \delta t \left( F_{x,I,jk}^{n,k} + \frac{H_{I+\frac{1}{2},jk}^{n,k} - H_{I,jk}^{n,k-1}}{\delta x/2} \right) \quad (4.74)$$



Very loosely, this converges according to

$$\left| u_{I,jk}^{*,k} - u_{I,jk}^* \right| \approx \left| \frac{u_{I,jk}^{*,k-1} - u_{I,jk}^*}{2} \right| \quad (4.75)$$

This iterative process continues until  $\left| u_{I,jk}^{*,k} - u_{I,jk}^* \right|$  falls below a specified tolerance. By default, the tolerance is  $\delta x/2$ . For the corrector step, this procedure is the same, except the boundary condition for the pressure term is:

$$\text{BXF}(J, K) \equiv H_{I+\frac{1}{2},jk}^{*,k} = \frac{H_{I,jk}^{*,k-1} + H_{I+1,jk}^{*,k-1}}{2} + \frac{\delta x}{2\delta t} (u_{I,jk}^{n+1,k-1} - u_{I,jk}^{n+1}) \quad (4.76)$$

**Case 3:** FDS uses a simple, direct-forcing immersed boundary method (IBM) [40] for block Cartesian geometries. Internal solid obstructions are represented as masked grid cells, but the no-flux condition (4.69) cannot be directly prescribed at the boundaries of these blocked cells. However, by solving the pressure equation several times within a time step, the normal component of velocity can be driven to within some specified tolerance of the desired value. At the start of a time step, the components of  $\mathbf{F}$  are computed at all cell faces that do not correspond to walls. At those cell faces that do correspond to

solid walls but are not located at the exterior of the computational grid, we prescribe (for example at the cell face where  $u_{ijk}$  “lives”):

$$F_{x,ijk}^{n,k} = -\frac{H_{i+1,jk}^{n,k-1} - H_{ijk}^{n,k-1}}{\delta x} - \frac{u_{ijk}^* - u_{ijk}^n}{\delta t} \quad (4.77)$$

at the predictor step, and

$$F_{x,ijk}^{*,k} = -\frac{H_{i+1,jk}^{*,k-1} - H_{ijk}^{*,k-1}}{\delta x} - \frac{u_{ijk}^{n+1} - \frac{1}{2}(u_{ijk}^* + u_{ijk}^n)}{\delta t/2} \quad (4.78)$$

at the corrector step. Note that the superscript  $n$  refers to the time step and  $k$  refers to the iteration of the pressure solver. Note that  $u_{ijk}^*$  and  $u_{ijk}^{n+1}$  are approximate because the true value of the velocity time derivative depends on the solution of the pressure equation, but since the most recent estimate of pressure is used, the approximation is fairly good. Also, even though there are small errors in normal velocity at solid surfaces, the divergence of each blocked cell remains exactly zero for the duration of the calculation. In other words, the total flux into a given obstruction is always identically zero, and the error in normal velocity is usually at least several orders of magnitude smaller than the characteristic flow velocity. When implemented as part of a predictor-corrector updating scheme, the no-flux condition at solid surfaces is maintained fairly well.

## Boundary Conditions at Mesh Interfaces

Dirichlet boundary conditions are applied at the interface between two meshes, which means that  $H$ , rather than  $\nabla H$ , is specified at the interface. Consider the interface between two non-overlapping meshes that abut at a common  $x$  boundary. The value of  $H$  in the center of the rightmost cell of the left mesh is denoted  $H_{I,jk}$ , and the value of  $H$  in the center of the leftmost cell of the right mesh is  $H_{1,jk}$ . The interface value of the left mesh is denoted  $H_{I+\frac{1}{2},jk}$ , and the interface value of the right mesh is denoted  $H_{\frac{1}{2},jk}$ .

At the start of a given time step  $n$ , the normal component of velocity at the interface of each respective mesh,  $\bar{u}_{I,jk}^n$  and  $\bar{u}_{0,jk}^n$ , are forced to take on the same value, their average from the previous time step. In the predictor phase of the time step, the values are updated independently to provide a first-order estimate at the next time step:

$$u_{I,jk}^{*,m} = \bar{u}_{I,jk}^n - \delta t \left( F_{x,I,jk}^n + \frac{H_{I+\frac{1}{2},jk}^m - H_{I,jk}^m}{\delta x/2} \right) ; \quad u_{I,jk}^* = \lim_{m \rightarrow \infty} u_{I,jk}^{*,m} \quad (4.79)$$

$$u_{0,jk}^{*,m} = \bar{u}_{0,jk}^n - \delta t \left( F_{x,0,jk}^n + \frac{H_{1,jk}^m - H_{\frac{1}{2},jk}^m}{\delta x/2} \right) ; \quad u_{0,jk}^* = \lim_{m \rightarrow \infty} u_{0,jk}^{*,m} \quad (4.80)$$

Ideally, these estimated values,  $u_{I,jk}^{*,m}$  and  $u_{0,jk}^{*,m}$ , should be the same, but they are typically not because the respective pressure fields are only guaranteed to be continuous at the interface, not differentiable (smooth). The common value of the pressure field at the interface boundary is:

$$H_{\frac{1}{2},jk}^m \equiv H_{I+\frac{1}{2},jk}^m = \frac{H_{I,jk}^{m-1} + H_{1,jk}^{m-1}}{2} + \frac{\delta x}{4\delta t} (u_{I,jk}^{*,m-1} - u_{0,jk}^{*,m-1}) \quad (4.81)$$

The purpose of the second term on the right hand side of Eq. (4.81) is seen by substituting  $H_{I+\frac{1}{2},jk}^m$  from Eq. (4.81) into Eq. (4.79) which leads to:

$$u_{I,jk}^{*,m} = \bar{u}_{I,jk}^n - \frac{u_{I,jk}^{*,m-1} - u_{0,jk}^{*,m-1}}{2} - \delta t \left( F_{x,I,jk}^n + \frac{(H_{I,jk}^{m-1} + H_{1,jk}^{m-1})/2 - H_{I,jk}^m}{\delta x/2} \right) \quad (4.82)$$

The extra term on the right hand side drives  $u_{I,jk}^{*,m}$  halfway towards  $u_{0,jk}^{*,m}$ , and vice versa. This iterative process continues until  $|u_{I,jk}^{*,m} - u_{0,jk}^{*,m}|$  falls below a specified tolerance. By default, the tolerance is  $\delta x/2$ .

For the corrector phase of the time step, the procedure is the same, except the boundary condition for the pressure term is:

$$H_{\frac{1}{2},jk}^m \equiv H_{I+\frac{1}{2},jk}^m = \frac{H_{I,jk}^{m-1} + H_{1,jk}^{m-1}}{2} + \frac{\delta x}{2\delta t} \left( u_{I,jk}^{n+1,m-1} - u_{0,jk}^{n+1,m-1} \right) ; \quad u_{I,jk}^{n+1} = \lim_{m \rightarrow \infty} u_{I,jk}^{n+1,m} \quad (4.83)$$

The factor of  $1/2$  instead of  $1/4$  on the last term is due to the structure of the corrector velocity step:

$$u_{I,jk}^{n+1,m} = \frac{\bar{u}_{I,jk}^n + \bar{u}_{I,jk}^*}{2} - \frac{u_{I,jk}^{n+1,m-1} - u_{0,jk}^{n+1,m-1}}{2} - \frac{\delta t}{2} \left( F_{x,I,jk}^* + \frac{(H_{I,jk}^{m-1} + H_{1,jk}^{m-1})/2 - H_{I,jk}^m}{\delta x/2} \right) \quad (4.84)$$

Note that throughout this iterative procedure, MPI calls are made to exchange values of  $u$  and  $H$  in the neighboring cells.

#### 4.3.4 Iterative Procedure for Updating Velocity

The Poisson solver in FDS produces an exact solution of Eq. (4.63) on each mesh. There are three problems with this solution, however:

1. The solution,  $H$ , is continuous at mesh interfaces, but the finite-difference of its gradient is not. This means that the normal component of velocity at the mesh interface will not agree at the next time step.
2. At solid internal boundaries, the normal component of velocity is not exactly zero because the normal component of  $\mathbf{F}$  is set equal to the previous value of the gradient of  $H$ . The no-flux boundary condition is only exact at external boundaries.
3. The perturbation pressure,  $\tilde{p}$ , that is included in  $\mathbf{F}$  is from the previous time step. Thus, after solving the Poisson equation, the value of  $\tilde{p}$  implicit in  $H$  will not equal the value in  $\mathbf{F}$ .

One solution to these three problems is to solve the Poisson equation multiple times, each time updating the lagged value of pressure until the normal component of velocity at internal solids and mesh boundaries converges within a specified tolerance, and until the old and new values of the perturbation pressure,  $\tilde{p}$ , converge to within a specified tolerance.

Following is a step by step procedure for advancing the velocity components. This same procedure is followed, with a few noted exceptions, in both the predictor and corrector stages of the time step.

1. The overlapping normal components of velocity that co-exist at the mesh interface are replaced by their average. Consider two meshes joined side by side in the  $x$  direction. The component  $u_I \equiv u_{I,jk}$  lives on the right boundary of the left hand mesh, and  $u_0 \equiv u_{0,jk}$  lives on the left boundary of the right hand mesh. Define the discrete “patch-averaged” field  $\bar{u}$  which is identical at all overlapping mesh points. To do this we simply average the coincident values of the normal velocity component at the mesh interfaces. For instance, considering the same side-by-side meshes as before,

$$\bar{u}_I = \bar{u}_0 \equiv \frac{1}{2} (u_{I,jk} + u_{0,jk}) \quad (4.85)$$

for all patch boundary cells  $j$  and  $k$ . Here, for simplicity, we are only considering the case in which the cell sizes are equivalent for the adjoining meshes (coarse-fine mesh interfaces are possible).

2. Compute  $\mathbf{F}_A(\bar{\mathbf{u}})$  as described in Sec. 4.3.2.
3. Add the baroclinic term  $\mathbf{F}(\bar{\mathbf{u}}) = \mathbf{F}_A(\bar{\mathbf{u}}) + \mathbf{F}_B(\bar{\mathbf{u}})$  as described in Sec. 4.3.2.
4. Compute the normal component of  $\mathbf{F}$  at solid surfaces from Eq. (4.77) or Eq. (4.78).
5. Solve the Poisson equation for the pressure  $H$  as described in Sec. 4.3.3. At the predictor stage:

$$\nabla^2 H^n = - \left( \frac{\nabla \cdot \mathbf{u}^* - \nabla \cdot \mathbf{u}^n - \nabla \cdot (\bar{\mathbf{u}}^n - \mathbf{u}^n)}{\delta t} \right) - \mathbf{F}(\bar{\mathbf{u}}^n) \quad (4.86)$$

At the corrector stage:

$$\nabla^2 H^* = - \left( \frac{2\nabla \cdot \mathbf{u}^{n+1} - \nabla \cdot \mathbf{u}^* - \nabla \cdot (\bar{\mathbf{u}}^* - \mathbf{u}^*) - \nabla \cdot \mathbf{u}^n - \nabla \cdot (\bar{\mathbf{u}}^n - \mathbf{u}^n)}{\delta t} \right) - \mathbf{F}(\bar{\mathbf{u}}^*) \quad (4.87)$$

The extra terms in the time derivative,  $\nabla \cdot (\bar{\mathbf{u}}^n - \mathbf{u}^n)$  and  $\nabla \cdot (\bar{\mathbf{u}}^* - \mathbf{u}^*)$ , “correct” the divergence error. The benefit to averaging the normal components of velocity at mesh interfaces is that  $\mathbf{F}$  is the same on each side of the interface, since all force terms are determined using the patch-averaged field. This also means that stress tensors computed at a mesh interface (which are buried in  $\mathbf{F}$ ) are symmetric; this symmetry is a requirement for angular momentum conservation. Thus, the patch-averaging procedure prevents the production of spurious vorticity at mesh interfaces.

6. Estimate the velocity field at the next time step. For the predictor step:

$$\mathbf{u}^{*,k} = \bar{\mathbf{u}}^n - \delta t \left( \mathbf{F}(\bar{\mathbf{u}}^n) + \nabla H^{k-1} \right) \quad (4.88)$$

At the corrector step:

$$\mathbf{u}^{n+1,k} = \frac{1}{2} \left( \bar{\mathbf{u}}^n + \bar{\mathbf{u}}^* - \delta t \left( \mathbf{F}(\bar{\mathbf{u}}^*) + \nabla H^{k-1} \right) \right) \quad (4.89)$$

Note that for both stages, the normal components of velocity at the interface are no longer expected to match because the individual pressure fields do not match exactly at the interface.

7. Check the convergence criteria. The default velocity tolerance is

$$|u_I^* - u_0^*| < 0.5 \delta x \quad (4.90)$$

and the default convergence criteria for the pressure,  $p = \rho(H - |\mathbf{u}|^2/2)$ , is

$$|\nabla \cdot (p^k - p^{k-1}) \nabla (1/\rho)| < 20/\delta x^2 \quad (4.91)$$

If the criteria are not met, return to Step 3.

This iterative scheme works well for most multiple mesh configurations, but it is excessively slow in the case of tunnel geometries, where potentially dozens of meshes might be aligned end to end to make up the tunnel. In such circumstances, there is an optional pre-conditioning scheme for the solution of the Poisson equation described in Appendix N.

## 4.4 Velocity Boundary Conditions

This section describes how the tangential component of velocity is specified at solid surfaces, mesh interfaces, or open boundaries to the outside atmosphere. The normal component of velocity is not specified directly at these boundaries, but rather indirectly via the pressure boundary condition, as described in Sec. 4.3.3.

### 4.4.1 Smooth Walls

In finite-volume LES, when the momentum equation is integrated over a cell adjacent to the wall it turns out that the most difficult term to handle is the viscous stress,  $\tau_w$ , because the wall-normal gradient of the streamwise velocity component cannot be resolved; the SGS stress at the wall is identically zero. We have, therefore, an entirely different situation than exists in the bulk flow at high Reynolds number where the viscous terms are negligible and the SGS stress is of critical importance. The fidelity of the SGS model still influences the wall stress, however, since other components of the SGS tensor affect the value of the near-wall velocity and hence the resulting viscous stress determined by the wall model. FDS models  $\tau_w$  with a logarithmic velocity profile [15] described below.

An important scaling quantity in the near-wall region is the friction velocity, defined as  $u_\tau \equiv \sqrt{\tau_w/\rho}$ . From the friction velocity we can define the nondimensional streamwise velocity  $u^+ \equiv u/u_\tau$  and nondimensional wall-normal distance  $y^+ \equiv y/\delta_v$ , where  $\delta_v = v/u_\tau = \mu/(\rho u_\tau)$  is the *viscous length scale*. In FDS, the law of the wall is approximated by

$$u^+ = y^+ \quad \text{for } y^+ < 11.81 \quad (4.92)$$

$$u^+ = \frac{1}{\kappa} \ln y^+ + B \quad \text{for } y^+ \geq 11.81 \quad (4.93)$$

where  $\kappa = 0.41$  is the von Kármán constant and  $B = 5.2$ . The region  $5 < y^+ < 30$ , where both viscous and inertial stresses are important, is referred to as the buffer layer. Following the work of Werner and Wengle [41], the solution in this region is approximated by matching the viscous region and log regions at  $y^+ = 11.81$ .

For the purposes of adapting the log law model to FDS we suppose that the first off-wall velocity component represents the profile sampled at a distance  $\delta y/2$  in the wall-normal direction—streamwise components of velocity are stored at the face center on a staggered grid. The density and molecular viscosity are taken as the average of the neighboring cell values and uniform on the cell face where the streamwise velocity component is stored.

### 4.4.2 Rough Walls

For rough walls FDS employs the log law presented in Pope [15],

$$u^+ = \frac{1}{\kappa} \ln \left( \frac{y}{s} \right) + \tilde{B}(s^+) \quad (4.94)$$

where  $s^+ = s/\delta_v$  is the roughness length in viscous units and  $s$  is the dimensional roughness. The distance to the wall,  $y$ , is taken as  $\delta y/2$  for the first off-wall grid cell. The parameter  $\tilde{B}$  varies with  $s^+$  but attains a constant value  $B_2 = 8.5$  in the fully rough limit. In FDS, we implement  $\tilde{B}$  as the following piece-wise function:

$$\tilde{B} = \begin{cases} B + (1/\kappa) \ln(s^+) & \text{for } s^+ < 5.83 \\ \tilde{B}_{\max} & \text{for } 5.83 \leq s^+ < 30.0 \\ B_2 & \text{for } s^+ \geq 30.0 \end{cases} \quad (4.95)$$

where  $\tilde{B}_{\max} = 9.5$ .

### 4.4.3 Near-Wall Eddy Viscosity Model

The WALE model (see Sec. 4.2.5) is used as the near-wall eddy viscosity model; that is, WALE is used to compute the eddy viscosity for any Cartesian cell adjacent to a solid boundary. There are two reasons for this. First, the test filtering operation needed for the subgrid kinetic energy in our implementation of Deardorff's model is not well defined near a wall. Second, with the WALE model the eddy viscosity goes to zero at the correct rate,  $v_t = O(y^3)$ , without the need for an explicit damping function.

### 4.4.4 Wall Damping of the Turbulent Viscosity

An alternative to the WALE model is to use Van Driest damping<sup>2</sup>. The turbulent viscosity  $v_t = \mu_t/\rho$  may be thought of as a “mixing length” squared divided by a time scale. For example, in the Smagorinsky model the mixing length is  $\ell_{\text{mix}} = C_s \Delta$  and the time scale is the inverse of the strain rate invariant  $1/|S|$ . Thus, the turbulent kinematic viscosity has units of length<sup>2</sup>/time.

To achieve the correct decay of the Reynolds stresses near a wall, Van Driest [42] proposed the following modification:

$$\ell_{\text{mix}} = C_s \Delta \left[ 1 - e^{-y^+/A} \right] \quad (4.96)$$

where  $A$  is a dimensionless empirical constant equal to 26. The term in brackets is referred to as the *Van Driest damping function*. In FDS, due to difficulties defining a consistent test filter for use with either the Deardorff or the dynamic Smagorinsky turbulence model near the wall, at corners, and inside cavities, the turbulent viscosity of the first off-wall cell is obtained from the Smagorinsky model with Van Driest damping applied to the mixing length as shown in Eq. (4.96) with  $y^+ = (\delta y/2)/\delta v$  and  $C_s = 0.2$ . See Sec. 4.4.1 for an explanation of terms. The viscosity near the wall is then given by

$$v_t = \ell_{\text{mix}}^2 |S| \quad (4.97)$$

where the strain rate,  $|S|$ , is defined in Eq. (4.16).

### 4.4.5 Open Boundaries (General, Wind)

An open boundary is where the fluid is allowed to enter or exit the computational domain based on local pressure gradients, like at an open window or door of a building. Typically, the gradients of the tangential components of velocity are set to zero at an open boundary. That is, the “ghost cell” values of the tangential velocity components are set equal to their values in the first grid cell. However, if the user specifies a wind field  $(u_\infty(z,t), v_\infty(z,t))$ , then the tangential components of velocity are set to their respective far-field values at all inflow boundaries. At outflow boundaries, the standard zero-gradient condition is applied.

### 4.4.6 Mesh Boundaries

At the interface between two meshes, the tangential components of velocity are taken directly from the neighboring mesh via an MPI exchange.

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<sup>2</sup>Van Driest damping was the default wall model in FDS Version 6

## 4.5 Time Step and Stability Constraints

In explicit schemes, stability criteria may often be understood in terms of using the time step to maintain physically realizable conditions. Below we examine the necessary conditions for stability in the presence of advection, diffusion, and expansion of the velocity and scalar fields.

### 4.5.1 The Courant-Friedrichs-Lowy (CFL) Constraint

The well-known CFL constraint given by

$$\text{CFL} = \delta t \frac{\|\mathbf{u}\|}{\delta x} \approx 1 \quad (4.98)$$

places a restriction on the time step due to the advection velocity. Physically, the constraint says that a fluid element should not traverse more than one cell within a time step. For LES, this constraint has the added advantage of keeping the implicit temporal and spatial filters consistent with each other. In other words, in order to resolve an eddy of size  $\delta x$ , the time step needs to be in concert with the CFL. If one were to employ an implicit scheme for the purpose of taking time steps say 10 times larger than the CFL limit, the smallest resolvable turbulent motions would then be roughly 10 times the grid spacing, which would severely limit the benefit of LES. In most cases, if one wishes the simulation to run faster, a better strategy is to coarsen the grid resolution while keeping the  $\text{CFL} \approx 1$ .

The exact CFL needed to maintain stability depends on the order (as well as other properties) of the time integration scheme and the choice of velocity norm. Three choices for velocity norm are available in FDS (set on MISC):

`CFL_VELOCITY_NORM=0` (corresponds to  $L_\infty$  norm of velocity vector)

$$\frac{\|\mathbf{u}\|}{\delta x} = \max \left( \frac{|u|}{\delta x}, \frac{|v|}{\delta y}, \frac{|w|}{\delta z} \right) + |\nabla \cdot \mathbf{u}| \quad (4.99)$$

`CFL_VELOCITY_NORM=1` (DNS and LES default, most restrictive, corresponds to  $L_1$  norm of velocity vector)

$$\frac{\|\mathbf{u}\|}{\delta x} = \frac{|u|}{\delta x} + \frac{|v|}{\delta y} + \frac{|w|}{\delta z} + |\nabla \cdot \mathbf{u}| \quad (4.100)$$

`CFL_VELOCITY_NORM=2` (VLES default,  $L_2$  norm of velocity vector)

$$\frac{\|\mathbf{u}\|}{\delta x} = \sqrt{(u/\delta x)^2 + (v/\delta y)^2 + (w/\delta z)^2} + |\nabla \cdot \mathbf{u}| \quad (4.101)$$

`CFL_VELOCITY_NORM=3` (SVLES default, least restrictive, corresponds to  $L_\infty$  norm of velocity vector)

$$\frac{\|\mathbf{u}\|}{\delta x} = \max \left( \frac{|u|}{\delta x}, \frac{|v|}{\delta y}, \frac{|w|}{\delta z} \right) \quad (4.102)$$

The addition of the magnitude of the velocity divergence to the velocity norm is discussed below in Sec. 4.5.3. Notice that `CFL_VELOCITY_NORM=3` omits this restriction and should therefore only be used for incompressible flows.

## 4.5.2 The Von Neumann Constraint

The Von Neumann constraint is given by

$$VN = \delta t \max [(\mu/\rho), D_\alpha] \sum_i \frac{1}{\delta x_i^2} < \frac{1}{2} \quad (4.103)$$

We can understand this constraint in a couple of different ways. First, we could consider the model for the diffusion velocity of species  $\alpha$  in direction  $i$ ,  $V_{\alpha,i} Y_\alpha = -D_\alpha \partial Y_\alpha / \partial x_i$ , and we would then see that VN is simply a CFL constraint due to diffusive transport.

We can also think of VN in terms of a total variation diminishing (TVD) constraint. That is, if we have variation (curvature) in the scalar field, we do not want to create spurious wiggles that can lead to an instability by overshooting the smoothing step. Consider the following explicit update of the heat equation for  $u$  in 1D. Here subscripts indicate grid indices and  $v$  is the diffusivity.

$$u_i^{n+1} = u_i^n + \frac{\delta t v}{\delta x_i^2} (u_{i-1}^n - 2u_i^n + u_{i+1}^n) \quad (4.104)$$

Very simply, notice that if  $\delta t v / \delta x^2 = 1/2$  then  $u_i^{n+1} = (u_{i-1}^n + u_{i+1}^n)/2$ . If the time step is any larger we overshoot the straight line connecting neighboring cell values. Of course, this restriction is only guaranteed to be TVD if the  $u$  field is “smooth”, else the neighboring cell values may be shifting in the opposite direction. Unfortunately, in LES there is no such guarantee and so the VN constraint can be particularly devilish in generating instabilities. For this reason, some practitioners like to employ implicit methods for the diffusive terms.

## 4.5.3 Realizable Mass Density Constraint

In an explicit Euler update of the continuity equation, if the time increment is too large the computational cell may be totally drained of mass, which of course is not physical. The constraint  $\rho^{n+1} > 0$  therefore leads to the following restriction on the time step:

$$\delta t < \frac{\rho^n}{\bar{\mathbf{u}}^n \cdot \nabla \rho^n + \rho^n \nabla \cdot \mathbf{u}^n} \quad (4.105)$$

We can argue that the case we are most concerned with is when  $\rho^n$  is near zero. A reasonable approximation to (4.105) then becomes (time location suppressed, summation over  $i$  is implied)

$$\delta t < \frac{\rho}{\bar{u}_i \left( \frac{\rho-0}{\delta x_i} \right) + \rho \nabla \cdot \mathbf{u}} = \left[ \frac{\bar{u}_i}{\delta x_i} + \nabla \cdot \mathbf{u} \right]^{-1} \quad (4.106)$$

Equation (4.106) basically adds the effect of thermal expansion to the CFL constraint and provides a reason to prefer `CFL_VELOCITY_NORM=1` as the basis for the time step restriction.

## 4.5.4 Realizable Fluid Volume Constraint

Mass conservation tells us that the time rate of change of a fluid element with mass  $\rho V$  does not change:

$$\frac{d(\rho V)}{dt} = 0. \quad (4.107)$$

Using continuity, Eq. (4.107) rearranges to

$$\nabla \cdot \mathbf{u} = \frac{1}{V} \frac{dV}{dt}, \quad (4.108)$$

where  $V(t)$  is the time-dependent volume of the fluid element. If  $\nabla \cdot \mathbf{u} < 0$ , the fluid element is under compression. In fire dynamics this usually occurs due to cooling (heat loss by radiation, for example). Equation (4.108) highlights the physical interpretation of the velocity divergence as the rate of volumetric expansion of the fluid *per unit volume*.

Equation (4.108) also implies a time step constraint. Consider an explicit update of Eq. (4.108) for the fluid volume:

$$V^{n+1} = V^n + \Delta t V^n (\nabla \cdot \mathbf{u})^n. \quad (4.109)$$

If the fluid element is in compression (the divergence is negative), positivity of the fluid volume requires the time step to be limited by

$$\Delta t < -(\nabla \cdot \mathbf{u})^{-1}. \quad (4.110)$$

Note that this is the analog of the positive mass density constraint when the divergence is positive and provides the rationale for using the absolute value of the divergence  $|\nabla \cdot \mathbf{u}|$  in the final version of the CFL constraint shown below.

#### 4.5.5 Heat Transfer Constraint

Note that the heat flux,  $\dot{q}_c''$ , has units of  $\text{W/m}^2$ . Thus, a velocity scale may be formed from  $(\dot{q}_c''/\rho_w)^{1/3}$ , where  $\rho_w$  is the gas phase density at the wall. Anytime we have a velocity scale to resolve, we have a CFL-type stability restriction. Therefore, the heat transfer stability check loops over all wall cells to ensure  $\delta t < (\delta x/2)/(\dot{q}_c''/\rho_w)^{1/3}$ . This check is an option. It is not done by default.

#### 4.5.6 Adjusting the Time Step

By default, the CFL is increased or decreased to remain between 0.8 and 1. To be clear, the CFL constraint is now given by

$$\text{CFL} = \delta t \frac{\|\mathbf{u}\|}{\delta x} \quad (4.111)$$

In DNS mode, the time step is also adjusted to maintain VN between 0.4 and 0.5. If either the CFL or VN is too large then the new time step is set to 90% of the allowable value. If both CFL and VN are below their minimum values then the current time step is increased by 10%. See the User's Guide [3] for details.

## Chapter 5

# Combustion (Chemically Reacting Flows)

The combustion model determines the mean chemical mass production rate of species  $\alpha$  per unit volume,  $\dot{m}_\alpha'''$ , in the species transport equation, Eq. (2.5). In general,  $\dot{m}_\alpha'''$  requires a closure model because the flame thickness is on the order of one millimeter while the grid spacing is typically on the order of tens of centimeters. This chapter describes a turbulent batch reactor model for  $\dot{m}_\alpha'''$  capable of handling a range of mixing conditions and chemical kinetics. In the non-premixed, fast chemistry limit, which is valid for the vast majority of FDS applications, the reactor model reduces to a simple “mixed is burnt” approximation called the Eddy Dissipation Concept (EDC) [43, 44].

The combustion model also determines the heat release rate per unit volume,  $\dot{q}'''$ , which is a quantity of fundamental importance in fire physics and typically the largest contribution to the velocity divergence, Eq. (3.26). Once  $\dot{m}_\alpha'''$  has been determined, the heat release rate follows by summing the mass production rates for each species times their respective heats of formation. Details are discussed below in Section 5.2.6.

Before discussing the combustion model, we first discuss of our *lumped species* approach (Sec. 5.1), which reduces the computational burden of the full chemical system by combining species into groups that transport and react together. In other words, we reduce the number of transport equations we need to solve, which significantly increases the speed of the computation.

In Sec. 5.2, we begin the discussion of our generalized combustion model which is designed to handle both fast and slow chemistry and a range of mixing conditions. For fire, this method holds the promise for improved prediction of carbon monoxide and soot. Each computational cell is treated as a partially-stirred batch reactor with a characteristic mixing time. Once reactants are mixed, the available reaction mechanisms range from infinitely fast chemistry to Arrhenius rate laws and reversible reactions. Our basic mixing-controlled, fast chemistry combustion model is presented in Sec. 5.2.4. Sec. 5.3 discusses several options for modeling extinction.

## 5.1 Lumped Species Approach

In the typical FDS problem the primitive species are lumped into reacting groups, and we consider the simple reaction



We refer to the Fuel, Air, and Products in Eq. (5.1) as *lumped species*. The lumped species approach is a simplified reaction progress variable approach [45] in which all the progress variables are mass fractions. This avoids any complications related to boundedness and ill-defined initial and boundary conditions.

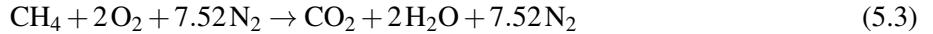
### 5.1.1 Relationship between Lumped and Primitive Species

In a simple hydrocarbon reaction, the reactants are the fuel, oxygen, and nitrogen and the products are carbon dioxide, water vapor, and nitrogen. For methane, the primitive species mass fractions are given by the composition vector

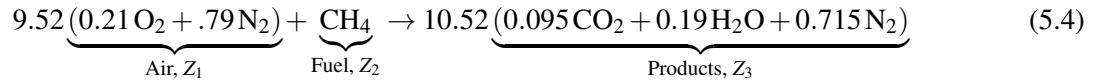
$$\mathbf{Y} = [Y_{\text{CH}_4} \ Y_{\text{O}_2} \ Y_{\text{N}_2} \ Y_{\text{CO}_2} \ Y_{\text{H}_2\text{O}}]^T \quad (5.2)$$

Lumped species are groups of primitive species which only exist in the flow in certain proportions. For example, Air can be assumed to be a lumped species composed of 21 % O<sub>2</sub>, 79 % N<sub>2</sub> by volume, plus trace amounts of water vapor and carbon dioxide. The key assumption made in lumping primitive species is that the new species groups transport (implying equal diffusivities) and react together.

In terms of primitive species, a one-step methane reaction may be written as



This is equivalent to



where 9.52 moles of Air react with 1 mole of Fuel to produce 10.52 moles of Products. Notice that the primitive species have been grouped by volume fraction into lumped species and the lumped species stoichiometric coefficients are the sum of the primitive species coefficients from Eq. (5.3). Note that  $9.52 \times 0.21$  is only approximately equal to 2. In practice the atom balance requires machine precision. To alleviate this issue, FDS internally normalizes the lumped species volume fractions and makes any necessary adjustments to the specified lumped stoichiometric coefficients. The lumped species mass fractions are denoted  $Z_i$ ,  $i = 1, \dots, N_Z$ , where  $N_Z$  is the number of tracked species. The first tracked species,  $Z_1$ , is typically the background species, Air.

The linear transformation from lumped species to primitive species is given by

$$\mathbf{Y} = A\mathbf{Z} \quad (5.5)$$

where  $A$  is the transformation matrix ( $N_y$  rows  $\times$   $N_z$  columns). Each column of  $A$  represents a different lumped species. The elements of  $A$  are the mass fractions for each primitive species in a given lumped species:

$$a_{\alpha i} = \frac{v_{\alpha i} W_{\alpha}}{\sum_{\beta} v_{\beta i} W_{\beta}} \quad (5.6)$$

where  $v_{\alpha i}$  are the volume fractions of primitive species  $\alpha$  in lumped species  $i$  and  $W_{\alpha}$  are the molecular weights. If we want the primitive species in Eq. (5.4) and, as an example, say we have  $\mathbf{Z} = [0.3 \ 0.2 \ 0.5]^T$ , we can transform from lumped species to primitive species via

$$\mathbf{Y} = \begin{bmatrix} Y_{\text{O}_2} \\ Y_{\text{N}_2} \\ Y_{\text{CH}_4} \\ Y_{\text{CO}_2} \\ Y_{\text{H}_2\text{O}} \end{bmatrix} = \begin{bmatrix} 0.2330 & 0 & 0 \\ 0.7670 & 0 & 0.7248 \\ 0 & 1 & 0 \\ 0 & 0 & 0.1514 \\ 0 & 0 & 0.1238 \end{bmatrix} \begin{bmatrix} 0.3 \\ 0.2 \\ 0.5 \end{bmatrix} = \begin{bmatrix} 0.0699 \\ 0.5925 \\ 0.2000 \\ 0.0757 \\ 0.0619 \end{bmatrix} \quad (5.7)$$

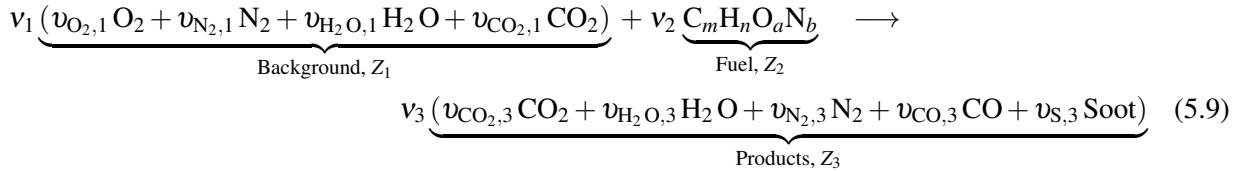
To transform back to lumped species from primitive species we can use:

$$\mathbf{Z} = B\mathbf{Y} \quad ; \quad B = (A^T A)^{-1} A^T \quad (5.8)$$

provided  $A$  has full rank and  $\mathbf{Y}$  is realizable (i.e., the forward transformation is also possible).

### 5.1.2 Default Hydrocarbon Combustion Chemistry

The default reaction equation in FDS, known as “simple chemistry,” is defined as follows:



Here, the volume fraction of primitive species  $\alpha$  in lumped species  $i$  is denoted by  $v_{\alpha i}$  and the stoichiometric coefficients for the lumped species  $i$  are denoted by  $v_i$ .

Carbon monoxide and soot yields are zero by default. The user can specify the CO and soot yields ( $y_{CO}$  and  $y_S$  respectively). The CO yield, and similarly for soot, is the mass of CO produced per mass of fuel reacted:

$$y_{CO} = \frac{\text{mass CO in Products}}{\text{mass of Fuel reacted}} \quad (5.10)$$

In this reaction system, Air (Background) is lumped species 1, Fuel is lumped species 2, and Products is lumped species 3. To find the stoichiometric coefficients of CO and soot within the products lumped species, FDS uses

$$v_2 v_{CO,2} = -v_1 \frac{W_1}{W_{CO}} y_{CO} \quad (5.11)$$

$$v_2 v_{S,2} = -v_1 \frac{W_1}{W_S} y_S \quad (5.12)$$

The remaining coefficients come from an atom balance.

**Example** Consider a methane–air reaction where methane has a specified CO yield of  $y_{CO} = 0.1$  and a Soot yield of  $y_S = 0.01$ . The default FDS reaction system lumps these species into Products. Note that, by default, Air is primarily composed of oxygen and nitrogen but includes trace amounts of carbon dioxide and water vapor. For this reaction the transformation matrix,  $A$ , is

	Air	Fuel	Products
CH <sub>4</sub>	0.000000	1.000000	0.000000
N <sub>2</sub>	0.762470	0.000000	0.719891
O <sub>2</sub>	0.230997	0.000000	0.000000
CO <sub>2</sub>	0.000591	0.000000	0.142935
CO	0.000000	0.000000	0.005584
H <sub>2</sub> O	0.005941	0.000000	0.131032
C	0.000000	0.000000	0.000558

The preceding table shows that the addition of carbon monoxide and soot increases the number of primitive species in the reaction from five to seven. The number of lumped species, however, remains at three—the composition of Products has changed to include the two additional species. Note that FDS prints the  $A$  matrix in the `CHID.out` file so that the user can double check the reaction system.

## 5.2 Turbulent Combustion

Modeling chemical reactions in turbulent flow is mathematically challenging because the length and time scales associated with the reactions may be orders of magnitude below what can be spatially and temporally resolved by the simulation. When the fuel and oxidizer are initially unmixed (diffusion flame) and the kinetics are fast compared with mixing, the simple Eddy Dissipation Concept (EDC) model [43, 44] is sufficient. However, for more complex reactions—such as carbon monoxide and soot formation—where reaction and mixing time scales may overlap, we require a more generalized approach.

To this end, we have developed a simple mixing environment method to close the mean chemical source term,  $\dot{m}_{\alpha}'''$ , in Eq. (2.5). For pure diffusion flames our method is similar to EDC, but the method is not limited to diffusion flames. Each computational cell is thought of as a *turbulent batch reactor*. At the start of a time step, each cell has an initial concentration of species (reactants, products, inerts) that exist with some degree of mixing. Generally, the rate of mixing is dominated by turbulence. The mixing time,  $\tau_{\text{mix}}$ , is discussed in Sec. 5.2.1. The mixing/reacting evolution equation and its numerical solution are described in Sections 5.2.2 and 5.2.3. Once mixed, species can react based on specified kinetic parameters—reactions may be infinitely fast (Sec. 5.2.4) or governed by an Arrhenius rate law (Sec. 5.2.5).

### 5.2.1 Reaction Time Scale Model

In this section we provide an expression for the mixing time based on the local state of the flow field. The basic idea behind the model we propose here is to consider the three physical processes of diffusion, subgrid-scale (SGS) advection, and buoyant acceleration and to take the fastest of these processes (locally) as the controlling flow time scale [46].

It is important to consider the behavior of an SGS model as the LES filter width (cell size) varies. The mixing times for diffusion, SGS advection, and buoyant acceleration scale differently with filter width and if we look to the limits of the filter scales an interesting picture emerges. Referring to Fig. 5.1, let us move from left to right along the horizontal axis following the thick black line which represents our time scale model for a hypothetical flow condition.

First, notice that the reaction time scale must be greater than or equal to the chemical time scale,  $\tau_{\text{chem}}$ , which is of the order of the traversal time of the flame thickness,  $\tau_{\text{chem}} \sim \delta/s_L$ , where  $\delta = D_F/s_L$ ,  $D_F$  is the diffusivity of the fuel, and  $s_L$  is the flame speed. At a slightly larger scale, we expect the mixing time to vary as the square of the filter width because the mixing is controlled by molecular diffusion. In this regime, denoted  $\tau_d$ , the numerical solution is a DNS. This scaling law is valid while  $\Delta$  is less than the Kolmogorov scale,  $\eta$ , the length scale of the smallest turbulent eddies (for this discussion we assume the Schmidt number ( $\text{Sc}$ ) is of order unity). For a sufficiently high Reynolds number flow (such that an inertial subrange exists), as the filter width increases beyond the Kolmogorov scale we encounter a regime, marked  $\tau_u$ , where turbulent advection controls the rate of mixing. Here the mixing time varies as the two-thirds power of the filter width [15]. This is the regime where most LES submodels are valid (It is important to appreciate that fire differs from turbulent combustion in that the assumption of locally high  $\text{Re}$  is frequently invalid).

Now, let us imagine what should happen to the mixing time as the filter width increases beyond the inertial subrange to a length scale larger than the height of the flame itself (actually a possibility in wildfire modeling). We would *not* expect the inertial range scaling to continue up through the so-called “energy-containing” range of turbulent length scales. Rather, for fires we expect buoyant acceleration to control the mixing at these relatively coarse scales. A time scale based on a constant acceleration goes as the square root of the filter width, as shown by the regime marked  $\tau_g$  in the diagram. This shift in scaling may appear minor given the log-log nature of the plot, but the effect of the acceleration-based time scale is indeed significant for large cell sizes. Finally, note that the flame height presents a limit to the reaction time scale, here denoted

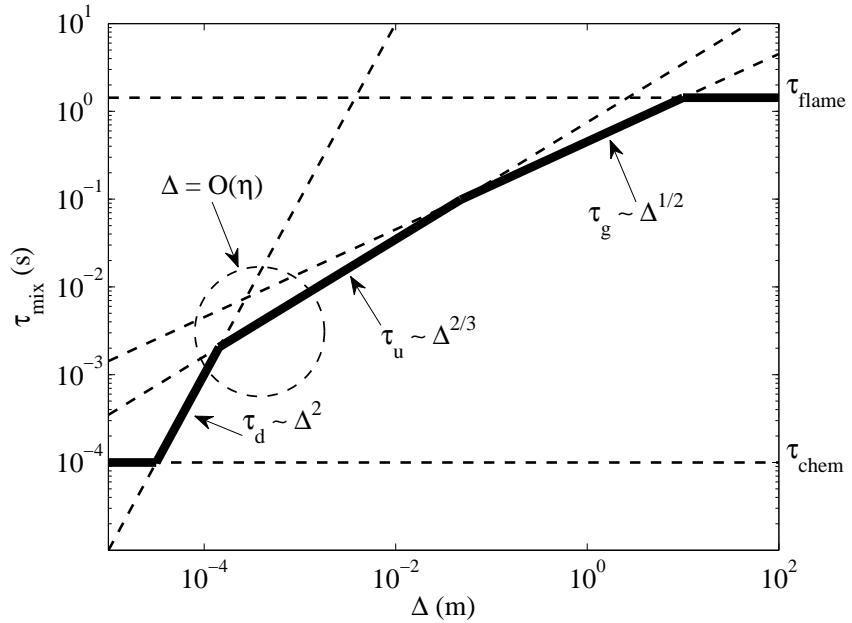


Figure 5.1: Reaction time scale model.

$\tau_{\text{flame}}$ , since all fuel must be consumed within a single cell.

Of course, the relative importance of the physical processes will depend on the flow. For example, if gravity is weak, the  $\tau_g$  line shifts up and may not affect the reaction time before the flame time scale is reached. If the flow is highly turbulent, the inertial range scaling may be more dominant, which would be indicated by a lowering of the  $\tau_u$  line. Or, for highly turbulent jet flames  $\tau_{\text{flame}}$  may be reached before the acceleration time scale has any effect. Perhaps more typical for low strain fires, if an inertial subrange does not exist (if the Reynolds number is too low relative to the Froude number), then the  $\tau_u$  line in Fig. 5.1 moves up out of the picture and we are left with diffusion and buoyancy to control the mixing.

The bold solid line in Fig. 5.1 is mathematically represented by

$$\tau_{\text{mix}} = \max(\tau_{\text{chem}}, \min(\tau_d, \tau_u, \tau_g, \tau_{\text{flame}})) \quad (5.13)$$

The mathematical details of the submodels are as follows:

$$\tau_d = \frac{\Delta^2}{D_F} \quad (5.14)$$

$$\tau_u = \frac{C_u \Delta}{\sqrt{(2/3)k_{\text{sgs}}}} \quad (5.15)$$

$$\tau_g = \sqrt{2\Delta/g} \quad (5.16)$$

where  $D_F$  is the diffusivity of the fuel species. Note that  $k_{\text{sgs}}$  is the unclosed subgrid kinetic energy per unit mass which by default is taken from the model for the turbulent viscosity (see Sec. 4.2, Eq. (4.17)). The advective time scale constant is calibrated to match the Heskestad flame height correlation [11] and is set to  $C_u = 0.4$ . The acceleration time scale  $\tau_g$  is the time required to travel a distance  $\Delta$  starting from rest under a constant acceleration,  $g = 9.81 \text{ m/s}^2$ .

## 5.2.2 Time Integration for Mixing and Reaction

Let the cell mean mass fraction of species  $\alpha$  be denoted as  $\tilde{Y}_\alpha(t)$ . Assume that any small parcel of gas within the cell exists in one of two states: completely unmixed or completely mixed. Let  $\hat{Y}_\alpha(t)$  denote the mass fraction of species  $\alpha$  in the mixed reactor zone, initially equal to the cell mean,  $\hat{Y}_\alpha(0) = \tilde{Y}_\alpha^0 \equiv \tilde{Y}_\alpha(0)$ . For convenience, we define  $\zeta(t)$  as the unmixed fraction of mass within the cell. As shown in Appendix E, the unmixed fraction evolves by the following simple ordinary differential equation (ODE):

$$\frac{d\zeta}{dt} = -\frac{\zeta}{\tau_{\text{mix}}} \quad (5.17)$$

with the solution

$$\zeta(t) = \zeta_0 e^{-t/\tau_{\text{mix}}} \quad (5.18)$$

The initial condition,  $\zeta_0$ , may be specified, modeled algebraically, or taken from the update of a passive scalar transport equation. Currently, FDS takes  $\zeta_0 = 1$  as default, an appropriate assumption for a turbulent diffusion flame.

At any point in time, the composition of the computational cell may be determined by combining the unmixed and mixed portions:

$$\tilde{Y}_\alpha(t) = \zeta(t) \tilde{Y}_\alpha^0 + (1 - \zeta(t)) \hat{Y}_\alpha(t) \quad (5.19)$$

Differentiating (5.19) in time and using (5.17) we see that our model for the chemical source term needed in (2.5) is given by

$$\dot{m}_\alpha''' = \rho \frac{d\tilde{Y}_\alpha}{dt} = \rho \left[ \frac{\zeta}{\tau_{\text{mix}}} (\hat{Y}_\alpha - \tilde{Y}_\alpha^0) + (1 - \zeta) \frac{d\hat{Y}_\alpha}{dt} \right] \quad (5.20)$$

Note that the unmixed fraction is comparable to the *complement* of the fraction of “reacting fine structures” in other EDC formulations [47, 48], but in our model this fraction evolves in time. A comparison between our model and that of Panjwani et al. [48] is developed further in Appendix F. An alternate derivation of (5.19) and (5.20) in terms of moments of the transport equation for the PDF is given in Appendix E. This derivation highlights that the implicit mixing model is a variant of the *interaction by exchange with the mean* or IEM model [49], which we refer to as the *interaction by exchange with the mixed mean* or IEMM.

Rather than work with the evolution equation (5.20) for the cell mean mass fraction,  $\tilde{Y}_\alpha(t)$ , it is more convenient to work with an analogous equation for the mass fraction within the mixed reactor zone,  $\hat{Y}_\alpha(t)$ . This equivalent evolution equation is derived as follows. It is assumed that the total mass within the grid cell is constant over a time step. The composition in the mixed reactor zone changes by two processes: mixing (mass is transferred from the unmixed zone to the mixed zone) and chemical reaction. We denote this mass by  $\rho V_c$ , where  $\rho$  is the initial cell mass density and  $V_c$  is the cell volume. The unmixed mass is denoted  $U(t)$  and the mixed mass is denoted  $M(t)$ . Given (5.18), the following equations describe the cell mass evolution:

$$\rho V_c = U(t) + M(t) \quad (5.21)$$

$$U(t) = \zeta(t) \rho V_c \quad (5.22)$$

$$M(t) = (1 - \zeta(t)) \rho V_c \quad (5.23)$$

Within the mixed reactor zone, let  $\hat{m}_\alpha(t)$  denote the mass of species  $\alpha$ . The mass fraction of  $\alpha$  in the mixed zone may then be written as

$$\hat{Y}_\alpha(t) \equiv \frac{\hat{m}_\alpha(t)}{M(t)} \quad (5.24)$$

This concentration is important because Arrhenius rate laws are based on the mixed composition only. The ODE governing the mixed species mass is

$$\begin{aligned}\frac{d\hat{m}_\alpha}{dt} &= \hat{Y}_\alpha \frac{dM}{dt} + M \frac{d\hat{Y}_\alpha}{dt} \\ &= -\tilde{Y}_\alpha^0 \frac{dU}{dt} + M \frac{d\hat{Y}_\alpha}{dt} \\ &= \rho V_c \left[ \frac{\zeta \tilde{Y}_\alpha^0}{\tau_{\text{mix}}} + (1 - \zeta) \frac{d\hat{Y}_\alpha}{dt} \right]\end{aligned}\quad (5.25)$$

The first term on the RHS accounts for mixing. The second term represents chemical kinetics. Note that in the second step we have utilized the fact that the unmixed composition remains constant (at the initial cell mean) throughout the time step. The third step follows from (5.17), (5.22), and (5.23).

### 5.2.3 Numerical Solution via Time-Splitting

In this section, we discuss the numerical solution of (5.25). Let  $\Delta t^k$  represent the  $k$ th sub-time step in the integration (less than or equal to the LES time step  $\delta t$ );  $t^k = 0$  at the start of the reactor integration. The integration is time split such that *mixing is done first, followed by reaction*. A simple explicit update of (5.25) over the sub-time interval  $t^k$  to  $t^k + \Delta t^k$  is given by

$$\hat{m}_\alpha^* = \hat{m}_\alpha(t^k) - [\zeta(t^k + \Delta t^k) - \zeta(t^k)] \tilde{Y}_\alpha^0 \rho V_c \quad (5.26)$$

$$\hat{Y}_\alpha^* = \hat{m}_\alpha^*/M^* \quad (5.27)$$

$$\hat{Y}_\alpha(t^k + \Delta t^k) = \hat{Y}_\alpha^* + \Delta \hat{Y}_\alpha^* \quad (5.28)$$

The superscript \* indicates a post-mixing value. The first step, (5.26), is an analytical solution for the mixing step (first term in (5.25)), obtained using (5.18). The mixing time scale  $\tau_{\text{mix}}$ , needed in (5.18), is computed once per LES time step using (5.13) and held constant during the reactor integration. The mixed mass,  $M^* = M(t^k + \Delta t^k)$ , is evaluated at the end of the subinterval using (5.23). For fast chemistry, we take only one sub-step ( $\Delta t^{k=1} = \delta t$ ). At the end of the time integration, the mixed zone composition,  $\hat{Y}_\alpha(\delta t)$ , is combined with the unmixed mass to obtain the final cell composition.

The method to determine  $\Delta \hat{Y}_\alpha^*$  (the change in mass fraction of  $\alpha$  due to chemical reaction) in (5.28) depends on the complexity of the reaction system. In the next section, we first discuss the simplest case of infinitely fast chemistry. Then we discuss finite-rate chemistry.

### 5.2.4 Infinitely Fast Chemistry

#### Single Reaction

For a single reaction, the change in fuel is based on the limiting reactant [44]:

$$\Delta \hat{Y}_F = -\min \left( \hat{Y}_F, \hat{Y}_\alpha \frac{v_F W_F}{v_\alpha W_\alpha} \right) \quad ; \quad \text{for all reactants, } \alpha \quad (5.29)$$

The minimum is taken in Eq. (5.29) to ensure that the reactant species mass fractions remain realizable.

This approach is used for the vast majority of large-scale fire applications—the so-called “mixed is burnt” assumption, where the mean chemical source term for the Fuel, F, is modeled using the Eddy Dissipation Concept (EDC) of Magnussen and Hjertager [43, 44]:

$$\dot{m}_F''' = -\rho \frac{\min(Y_F, Y_A/s)}{\tau_{\text{mix}}} \quad (5.30)$$

Here,  $Y_F$  and  $Y_A$  are the cell mean mass fractions of Fuel and Air, respectively, and  $s$  is the mass stoichiometric coefficient for Air.

## Multiple Reactions

For multiple reactions with infinitely fast chemistry, the reaction rate is treated as a second-order (assuming two reactants) Arrhenius reaction with zero activation energy. The Arrhenius constant is set to a large value. Further discussion of the time integration for these cases is given in Appendix D.

### 5.2.5 Finite-Rate Chemistry (Arrhenius Reaction)

Consider a simple one-step forward reaction:



The rate expression for species A with a mixed zone concentration of  $C_A$  in mol/cm<sup>3</sup> and rate constant  $k$  is

$$\frac{dC_A}{dt} = -k C_A^a C_B^b \quad (5.32)$$

Consider a set of  $N_r$  reactions with fuel F. The reaction rate (mol/(cm<sup>3</sup> s)) for F in the  $i$ th reaction is

$$r_{F,i} = -k_i \prod C_\alpha^{a_{\alpha,i}} \quad (5.33)$$

For the  $i$ th Arrhenius reaction, the rate constant,  $k_i$ , depends on the temperature,  $T$ , the temperature exponent,  $n_i$ , the pre-exponential factor,  $A_i$ , and the activation energy,  $E_i$ :

$$k_i = A_i T^{n_i} e^{-E_{a,i}/RT} \quad (5.34)$$

Note that the units of  $E_a$  are J/mol and units of  $A$  are ((mol/cm<sup>3</sup>)<sup>1- $\sum a_\alpha$ ) / s. Note that  $\sum a_\alpha$  is the *order* of the reaction. The units of  $A$  take the appropriate form to ensure the units of Eq. (5.33) are mol/(cm<sup>3</sup> s).</sup>

The reaction rate for species  $\alpha$  of the  $i$ th reaction is based on the ratio of stoichiometric coefficients:

$$r_{\alpha,i} = \left( \frac{v_{\alpha,i}}{v_{F,i}} \right) r_{F,i} \quad (5.35)$$

The change in concentration for species  $\alpha$  within the mixed reactor zone is then:

$$\frac{dC_\alpha}{dt} = \sum_i r_{\alpha,i} \quad (5.36)$$

FDS only transports lumped species and only lumped species can be consumed or created. Note, however, that any of the primitive species may participate in a reaction rate law.

It is more convenient for FDS to work in terms of mass fractions,  $Y_\alpha$ . The concentrations (mol –  $\alpha$ /cm<sup>3</sup>) and mass fractions (kg- $\alpha$ /kg) are related by  $C_\alpha = Y_\alpha \rho / (W_\alpha \times 1000)$ , where the density,  $\rho$ , has units of

$\text{kg}/\text{m}^3$ . To simplify the calculations within FDS, density and molecular weight are pulled out of the product concentrations on the right hand side of Eq. (5.33) and combined with the other constants to form  $A'$ :

$$A'_i = A_i \left( \prod [W_\alpha \times 1000]^{-a_{\alpha,i}} \right) \times \left( \frac{1 \text{ kmol}}{10^3 \text{ mol}} \right) \times \left( \frac{10^6 \text{ cm}^3}{1 \text{ m}^3} \right) \times W_F \quad (5.37)$$

Using  $A'_i$ , the reaction rate in mass units becomes

$$r'_{F,i} = -A'_i \rho^{\sum a_{\alpha,i}} T^{n_i} e^{-E_i/RT} \prod Y_\alpha^{a_{\alpha,i}} [=] \left( \frac{\text{kg F}}{\text{m}^3 \text{s}} \right) \quad (5.38)$$

The mass rate of reaction per unit volume for species  $\alpha$  in the  $i$ th reaction is

$$r'_{\alpha,i} = \left( \frac{v_{\alpha,i} W_\alpha}{v_{F,i} W_F} \right) r'_{F,i} [=] \left( \frac{\text{kg } \alpha}{\text{m}^3 \text{s}} \right) \quad (5.39)$$

Last, the rate of change in composition for species  $\alpha$  in the mixed reactor zone becomes

$$\frac{d\hat{Y}_\alpha}{dt} = \frac{1}{\rho} \sum_i r'_{\alpha,i} [=] \left( \frac{\text{kg } \alpha}{\text{kg s}} \right) \quad (5.40)$$

See Appendix D for a discussion on the algorithm used to ensure species remain bounded.

### Third Body Reactions

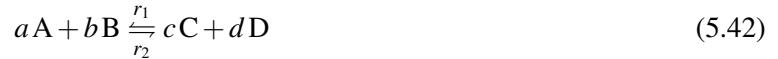
At low pressures, it is common to see so-called *third body* reactions. These reactions require the presence of some other molecule, M, for heat dissipation [50]. The reaction scheme is usually written as



The Arrhenius rate law has a first-order dependence on the concentration of M. Since M may be *any* other molecule, we take  $C_M = \bar{p}/(RT) = \rho/(\bar{W} \times 1000) [=] \text{ mol/cm}^3$ . Some molecules may be more or less efficient as a collision species. This non-unity effect may be specified in the input file. If non-unity species are present, then the concentration of M is taken as the sum of all the individual species concentrations times their weighting factor (if all species have a weighting factor of 1, then the result is  $C_M$ ).

### Equilibrium Chemistry

Consider the reversible chemical reaction



At equilibrium, the forward and reverse reaction rates are equal,  $r_1 = r_2$ . The rate constants for each direction are the  $k$ s in Eq. (5.32). The equilibrium constant is defined as  $K \equiv k_1/k_2$ . The equilibrium constant is computed via

$$K(T) = C_M^{v_n} \exp(-\Delta G_{rxn}^\circ(T)/(RT)) \quad (5.43)$$

where  $v_n$  is the net reaction order (for Eq. 5.42  $v_n = a + b - c - d$ ) and

$$G_{rxn}^\circ = dG_D^\circ + cG_C^\circ - bG_B^\circ - aG_A^\circ \quad (5.44)$$

Note that the Gibbs free energy as a function of temperature is defined for any FDS database that uses NASA polynomials. Gibbs free energy can be specified for user defined species.

## Time Integration for Finite-Rate Chemistry

For reactions other than single step, mixing controlled chemistry, a fourth-order explicit integrator with error control is used. The time integration follows the procedure outlined in Eqs. (5.26) and (5.28), but multiple subiterations are generally needed. The change in composition over the subinterval in the mixed reactor zone,  $\Delta\hat{Y}_\alpha^*$ , is usually obtained by integrating an Arrhenius rate law (we say “usually” because a combination of fast and finite-rate chemistry is permissible). More detail on the numerical methods of the integrator, including a method to combat stiff chemistry, can be found in Appendix D.

### 5.2.6 Heat Release Rate

At the end of the LES/DNS time step,  $\delta t$ , the mixed reactor zone mass fraction of species alpha is computed using the change in fuel concentration for each of the  $N_r$  reactions:

$$\hat{Y}_\alpha(\delta t) = \tilde{Y}_\alpha^0 + \sum_{i=1}^{N_r} \left( \frac{v_{\alpha,i} W_\alpha}{v_{F,i} W_{F,i}} \right) \Delta\hat{Y}_{F,i} \quad (5.45)$$

The new cell mean mass fraction is found by evaluating Eq. (5.19) at the end of the time step:

$$\tilde{Y}_\alpha(\delta t) = \zeta(\delta t) \tilde{Y}_\alpha^0 + (1 - \zeta(\delta t)) \hat{Y}_\alpha(\delta t) \quad (5.46)$$

The heat release per unit volume, defined as  $\dot{q}''' \equiv -\sum_\alpha \dot{m}_\alpha''' \Delta h_{f,\alpha}^0$ , is then found by summing the species mass production rates times the respective heats of formation:

$$\dot{q}''' = -\rho \sum_\alpha \frac{(\tilde{Y}_\alpha(\delta t) - \tilde{Y}_\alpha^0)}{\delta t} \Delta h_{f,\alpha}^0 \quad (5.47)$$

## 5.3 Extinction

A limitation of the default mixing-controlled reaction model described above is that it assumes fuel and oxygen always react regardless of the local temperature, reactant concentration, or strain rate. For large-scale, well-ventilated fires, this approximation is usually sufficient. However, if a fire is in an under-ventilated compartment, or if a suppression agent like water mist or CO<sub>2</sub> is introduced, or if the strain between the fuel and oxidizing streams is high, burning may not occur.

FDS uses simple empirical rules—which ignore strain—to predict local extinction within a given grid cell based on resolved species concentrations and the mean cell temperature. There are two options for predicting extinction, both of which are based on the concept of a *critical flame temperature*. The basic theory behind the critical flame temperature is described in Beyler's chapter of the SFPE Handbook [51] even though he never explicitly uses this term. In fact, it is mentioned only a few times in the Handbook, and each time it has a slightly different interpretation.

### 5.3.1 Critical Flame Temperature

A diffusion flame immersed in a vitiated atmosphere will extinguish before consuming all the available oxygen. The classic example of this behavior is a candle burning within an inverted jar. With this picture in mind, consider a control volume characterized by a bulk temperature,  $T$ , a mass,  $m$ , and an oxygen mass fraction,  $Y_{\text{O}_2}$ . Complete combustion of the oxygen within the control volume would release a quantity of energy given by:

$$Q = m Y_{\text{O}_2} \left( \frac{\Delta H}{r_{\text{O}_2}} \right) \quad (5.48)$$

where  $\Delta H/r_{\text{O}_2}$  has a relatively constant value of approximately 13100 kJ/kg for most fuels of interest in fire applications [52]. Under adiabatic conditions, the energy released by the reaction of the available oxygen and the corresponding stoichiometric amount of fuel would increase the bulk temperature of the gases to  $T_f$ :

$$Q = m \bar{c}_p (T_f - T) \quad (5.49)$$

The average specific heat of the gases can be calculated based on the composition of the combustion products as:

$$\bar{c}_p = \frac{1}{(T_f - T)} \sum_{\alpha} \int_T^{T_f} Y_{\alpha} c_{p,\alpha}(T') dT' \quad (5.50)$$

Equating Eqs. (5.48) and (5.49) yields:

$$Y_{\text{O}_2} = \frac{\bar{c}_p (T_f - T)}{\Delta H / r_{\text{O}_2}} \quad (5.51)$$

The critical flame temperature (CFT), as interpreted within FDS, is based on the *limiting oxygen index* (LOI), the oxygen volume fraction in the oxidant stream at the point of flame extinction. The adiabatic flame temperature corresponding to a stoichiometric mixture of fuel and oxygen at the LOI can be derived using Eq. (5.51):

$$T_{\text{OI}} = T_{\infty} + Y_{\text{OI}} \left( \frac{\Delta H / r_{\text{O}_2}}{\bar{c}_p} \right) ; \quad Y_{\text{OI}} = \frac{X_{\text{OI}} W_{\text{O}_2}}{X_{\text{OI}} W_{\text{O}_2} + (1 - X_{\text{OI}}) W_{\text{N}_2}} \quad (5.52)$$

Beyler's chapter of the SFPE Handbook [51] includes  $X_{\text{OI}}$  and  $T_{\text{OI}}$  for several common fuels. For other fuels, a default values are used. These values are given in the FDS User's Guide [3].

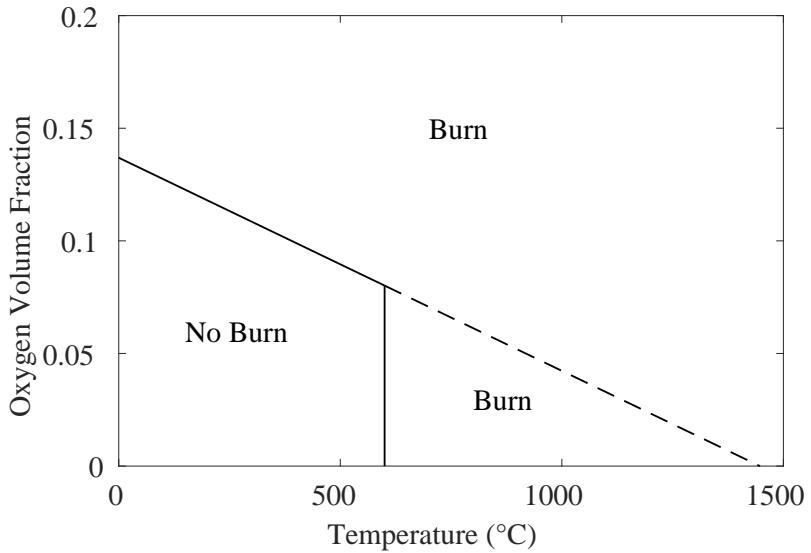


Figure 5.2: Extinction criteria for the 'EXTINCTION 1' model.

### 5.3.2 Extinction Based Mainly on Oxygen Concentration

The first of two optional extinction models (referred to as 'EXTINCTION 1' in the FDS input file) linearizes Eq. (5.51) to form a limiting oxygen concentration<sup>1</sup> that is a piecewise-linear function of the cell bulk temperature,  $T_{ijk}$  (see Fig. 5.2):

$$X_{O_2,\text{lim}}(T_{ijk}) = \begin{cases} X_{O_1} \left( \frac{T_{O_1} - T_{ijk}}{T_{O_1} - T_\infty} \right) & T_{ijk} < T_{fb} \\ 0 & T_{ijk} \geq T_{fb} \end{cases} \quad (5.53)$$

If  $X_{O_2,ijk} < X_{O_2,\text{lim}}$ , local extinction is assumed and  $\dot{m}_\alpha''' = 0$  and  $\dot{q}''' = 0$  for that grid cell at that time step. At an ambient temperature of 20 °C, the default limiting oxygen volume fraction is 0.135. This value is consistent with the measurements of Morehart et al. [53], who measured the oxygen concentration near self-extinguishing flames. They found that flames self-extinguished at oxygen volume fractions of 12.4 % to 14.3 %. Note that their results are expressed as volume, not mass, fractions. Beyler's chapter in the SFPE Handbook references other researchers who measured oxygen concentrations at extinction ranging from 12 % to 15 %.

The 'EXTINCTION 1' model is intended for relatively coarse fire simulations where the grid cell cannot resolve details of the flame structure or capture flame temperatures. The "free-burn" temperature,  $T_{fb}$ , in Eq. (5.53) is needed for simulations in which the characteristic grid cell size,  $\delta x$ , is much larger than 1 cm. In such cases, the combustion occurs within a fraction of the grid cell and its energy cannot raise the cell bulk temperature to the critical value. Its default value is 600 °C. Measurements of Pitts [54], Bundy [55], and others, have shown that the upper layer oxygen concentration drops to zero in flashover compartment fire experiments when the temperature increases above approximately 600 °C.

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<sup>1</sup>The extinction model is written in terms of the oxygen *volume* fraction,  $X_{O_2}$ , because usually experimental results are reported as such. However, within the numerical algorithm, all values are converted into mass fraction,  $Y_{O_2}$ .

### 5.3.3 Extinction Based on Both Fuel and Oxygen

The second optional extinction model in FDS, referred to as 'EXTINCTION 2', considers both the oxygen and the fuel content of a given grid cell at the start of a time step. If the potential heat release from the reactants cannot raise the temperature of the cell above the empirically determined critical flame temperature,  $T_{CFT}$ , combustion is suppressed. Consider the simple reaction  $\text{Fuel} + \text{Air} \rightarrow \text{Products}$ . The mass fractions of lumped species Fuel, Air, and Products in the mixed portion of the grid cell at the beginning and the end of the reaction part of the time step are  $[Z_F^0, Z_A^0, Z_P^0]$  and  $[Z_F, Z_A, Z_P]$ , respectively. The Products include the products of combustion as well as diluents like argon or water vapor from droplet evaporation. Define a modified form of the equivalence ratio:

$$\tilde{\phi} \equiv \min \left( 1, \frac{sZ_F^0}{Z_A^0} \right) = \frac{Z_A^0 - Z_A}{Z_A^0} \quad (5.54)$$

where  $s$  is the mass stoichiometric coefficient for Air (mass of Air required per mass of Fuel consumed). The extinction criterion assumes that excess fuel acts as a diluent, but excess air and a proportional amount of products do *not*. Hence, a large computational cell that is mostly air with a small amount of fuel is likely to burn, whereas a cell that is mostly fuel with little air will not. To achieve this, a fraction of the total mass of the grid cell equal to  $(1 - \tilde{\phi})(Z_A^0 + Z_P^0)$  has been removed from the calculation of the enthalpy. With this amount of mass removed, the extinction criterion is given by:

$$Z_F^0 h_F(T) + \tilde{\phi} Z_A^0 h_A(T) + \tilde{\phi} Z_P^0 h_P(T) < Z_F h_F(T_{CFT}) + [Z_P - (1 - \tilde{\phi})Z_P^0] h_P(T_{CFT}) \quad (5.55)$$

where  $T$  is the pre-reaction mean cell temperature and  $T_{CFT}$  is the critical flame temperature. Note that  $h_\alpha(T)$  represents the chemical plus sensible enthalpy; thus, the left-hand-side includes the combustion heat release. If the inequality, (5.55), holds, combustion is suppressed—the combustion heat release is not sufficient to raise the product mixture above its critical flame temperature. Notice that the right-hand side of the inequality contains no air because all excess air has been removed from consideration. In other words, for an infinitely fast reaction,  $Z_A - (1 - \tilde{\phi})Z_A^0 \equiv 0$ .

This extinction model can be applied to multiple reaction schemes. The critical flame temperature criterion is applied to the entire reaction. In other words, the individual reactions are allowed to occur, and the enthalpy inequality (5.55) is applied to the initial and final species mass fractions. If insufficient energy has been released, all reactions are suppressed and the species mass fractions are returned to their original values at the start of the time step.

If multiple fuels are present in the simulation, the critical flame temperature is taken as a weighted average of the CFTs of the multiple fuels that might be present within the grid cell at the start of the time step.

### 5.3.4 Auto-Ignition Temperature

As a convenience to users, FDS is designed so that there is no need to create an ignition source to initiate combustion—fuel and air burn on contact until the combustion becomes unviable as discussed above. However, in certain fire scenarios this assumption leads to spurious burning of fuel gas at the boundary of an oxygen-starved compartment. To prevent this, it is possible to turn off the assumption of piloted ignition by way of an auto-ignition temperature. If the cell temperature is below the user-specified auto-ignition temperature (AIT) for all fuels in the cell, combustion is suppressed. The auto-ignition temperature for each fuel is zero by default; thus, the user does not need to specify an ignition source when using the default combustion model because fuel and oxygen burn on contact.



# Chapter 6

## Thermal Radiation

Gas phase thermal conduction and radiation are represented by the divergence of the heat flux vector in the energy equation,  $\nabla \cdot \dot{\mathbf{q}}''$ . This chapter describes the equations that govern the radiative component,  $\dot{\mathbf{q}}_r''$ .

### 6.1 Radiation Transport Equation

The Radiative Transport Equation (RTE) for an absorbing, emitting, and scattering medium is [56]

$$\mathbf{s} \cdot \nabla I_\lambda(\mathbf{x}, \mathbf{s}) = \underbrace{-\kappa(\mathbf{x}, \lambda) I_\lambda(\mathbf{x}, \mathbf{s})}_{\text{Energy loss by absorption}} - \underbrace{\sigma_s(\mathbf{x}, \lambda) I_\lambda(\mathbf{x}, \mathbf{s})}_{\text{Energy loss by scattering}} + \underbrace{B(\mathbf{x}, \lambda)}_{\text{Emission source term}} + \underbrace{\frac{\sigma_s(\mathbf{x}, \lambda)}{4\pi} \int_{4\pi} \Phi(\mathbf{s}', \mathbf{s}) I_\lambda(\mathbf{x}, \mathbf{s}') d\mathbf{s}'}_{\text{In-scattering term}} \quad (6.1)$$

where  $I_\lambda(\mathbf{x}, \mathbf{s})$  is the radiation intensity at wavelength,  $\lambda$ ;  $\mathbf{s}$  is the direction vector of the intensity; and  $\kappa(\mathbf{x}, \lambda)$  and  $\sigma_s(\mathbf{x}, \lambda)$  are the local absorption and scattering coefficients, respectively.  $B(\mathbf{x}, \lambda)$  is the emission source term, describing how much heat is emitted by the local mixture of gas, soot and droplets/particles. The integral on the right hand side describes the in-scattering from other directions. The in-scattering and scattering terms are detailed in Sec. 6.3.

In practical simulations, the spectral dependence of the RTE cannot be resolved accurately. Instead, the radiation spectrum is divided into a relatively small number of bands and a separate RTE is derived for each band. For instance, the band specific RTE for a non-scattering gas is

$$\mathbf{s} \cdot \nabla I_n(\mathbf{x}, \mathbf{s}) = B_n(\mathbf{x}) - \kappa_n(\mathbf{x}) I_n(\mathbf{x}, \mathbf{s}), \quad n = 1 \dots N \quad (6.2)$$

where  $I_n$  is the intensity integrated over the band  $n$ , and  $\kappa_n$  is the appropriate mean absorption coefficient for the band. When the intensities corresponding to the bands are known, the total intensity is calculated by summing over all the bands

$$I(\mathbf{x}, \mathbf{s}) = \sum_{n=1}^N I_n(\mathbf{x}, \mathbf{s}) \quad (6.3)$$

#### 6.1.1 Radiation Source Term

The emission source term for radiation band  $n$  is

$$B_n(\mathbf{x}) = \kappa_n(\mathbf{x}) I_{b,n}(\mathbf{x}) \quad (6.4)$$

where  $I_{b,n}$  is the fraction of the blackbody radiation at temperature  $T(\mathbf{x})$ :

$$I_{b,n}(\mathbf{x}) = F_n(\lambda_{\min}, \lambda_{\max}) \sigma T(\mathbf{x})^4 / \pi \quad (6.5)$$

and  $\sigma$  is the Stefan-Boltzmann constant. The calculation of factors  $F_n$  is explained in Ref. [57]. The calculation of the absorption coefficients,  $\kappa_n$ , is discussed in Appendix C.

Even with a reasonably small number of bands, solving multiple RTEs is very time consuming. Fortunately, in most large-scale fire scenarios soot is the most important combustion product controlling the thermal radiation from the fire and hot smoke. As the radiation spectrum of soot is continuous, it is possible to assume that the gas behaves as a gray medium. The spectral dependence is then lumped into one absorption coefficient ( $N = 1$ ) and the source term is given by the blackbody radiation intensity [58]

$$I_b(\mathbf{x}) = \frac{\sigma T(\mathbf{x})^4}{\pi} \quad (6.6)$$

This is the default mode of FDS. For optically thin flames, however, where the yield of soot is small compared to the yields of CO<sub>2</sub> and water vapor, the gray gas assumption can lead to an over-prediction of the emitted radiation. From a series of numerical experiments using methane as the fuel, it has been found that six bands ( $N = 6$ ) provide an accurate representation of the most important radiation bands of the fuel, CO<sub>2</sub>, and water vapor [59]. Table 6.1 through Table 6.9 list the band limits for various fuel species. The location of the bands have been adjusted to accommodate most of the features of the fuels spectra. If the absorption of the fuel is known to be important, separate bands can be reserved for fuel, increasing the total number of bands,  $N$ . The number of additional bands depends on the fuel, as discussed in Appendix C.

### 6.1.2 Radiation Contribution to Energy Equation

The radiant heat flux vector  $\dot{\mathbf{q}}_r''$  is defined

$$\dot{\mathbf{q}}_r''(\mathbf{x}) = \int_{4\pi} I(\mathbf{x}, \mathbf{s}) \mathbf{s} d\Omega \quad (6.7)$$

The interaction between the electromagnetic radiation and the gas molecules and soot particles contributes to the gas phase energy equation through the source term which, under the gray gas assumption, can be written as

$$-\nabla \cdot \dot{\mathbf{q}}_r''(\mathbf{x})(\text{gas}) = \kappa(\mathbf{x}) [U(\mathbf{x}) - 4\pi I_b(\mathbf{x})] \quad ; \quad U(\mathbf{x}) = \int_{4\pi} I(\mathbf{x}, \mathbf{s}) d\Omega \quad (6.8)$$

This term then appears as a gas phase loss term in the radiation transfer equation. For  $N$  bands, the contribution to the radiative source term in the energy equation is

$$-\nabla \cdot \dot{\mathbf{q}}_r''(\mathbf{x})(\text{gas}) = \sum_{n=1}^N \kappa(\mathbf{x}) U_n(\mathbf{x}) - 4\pi B_n(\mathbf{x}) \quad ; \quad U_n(\mathbf{x}) = \int_{4\pi} I_n(\mathbf{x}, \mathbf{s}) d\Omega \quad (6.9)$$

In words, the net thermal energy gained by a grid cell from the radiation field is the difference between that which is absorbed and that which is emitted.

Table 6.1: Limits of the spectral bands for methane ( $\text{CH}_4$ ).

$\omega$ (1/cm)	10000	3800	3400	2400	2174	1000	50
6 Band Model	1	2	3	4	5	6	
Major Species	Soot $\text{CO}_2, \text{H}_2\text{O}$	$\text{CO}_2$ $\text{H}_2\text{O}, \text{Soot}$	$\text{CH}_4$ Soot	$\text{CO}_2$ Soot	$\text{H}_2\text{O}, \text{CH}_4$ Soot	$\text{H}_2\text{O}$ $\text{CO}_2$	
$\lambda$ ( $\mu\text{m}$ )	1.00	2.63	2.94	4.17	4.70	10.0	200

Table 6.2: Limits of the spectral bands for ethane ( $\text{C}_2\text{H}_6$ ).

$\omega$ (1/cm)	10000	3800	3350	2550	1650	1090	50
6 Band Model	1	2	3	4	5	6	
Major Species	Soot $\text{CO}_2, \text{H}_2\text{O}$	$\text{CO}_2$ $\text{H}_2\text{O}, \text{Soot}$	$\text{C}_2\text{H}_6$ Soot	$\text{CO}_2$ $\text{CO}, \text{H}_2\text{O}, \text{Soot}$	$\text{C}_2\text{H}_6$ $\text{H}_2\text{O}, \text{Soot}$	$\text{H}_2\text{O}$ $\text{CO}_2, \text{C}_2\text{H}_6$	
$\lambda$ ( $\mu\text{m}$ )	1.00	2.63	2.99	3.92	6.06	9.17	200

Table 6.3: Limits of the spectral bands for ethylene ( $\text{C}_2\text{H}_4$ ).

$\omega$ (1/cm)	10000	3800	3375	2800	1650	780	50
6 Band Model	1	2	3	4	5	6	
Major Species	Soot $\text{CO}_2, \text{H}_2\text{O}$	$\text{CO}_2$ $\text{H}_2\text{O}, \text{Soot}$	$\text{C}_2\text{H}_4$ Soot	$\text{CO}_2$ $\text{CO}, \text{H}_2\text{O}, \text{Soot}$	$\text{C}_2\text{H}_4$ $\text{H}_2\text{O}, \text{Soot}$	$\text{H}_2\text{O}$ $\text{CO}_2$	
$\lambda$ ( $\mu\text{m}$ )	1.00	2.63	2.96	3.57	6.06	12.82	200

Table 6.4: Limits of the spectral bands for propylene ( $\text{C}_3\text{H}_6$ ).

$\omega$ (1/cm)	10000	3800	3250	2600	1950	1175	50
6 Band Model	1	2	3	4	5	6	
Major Species	Soot $\text{CO}_2, \text{H}_2\text{O}$	$\text{CO}_2$ $\text{H}_2\text{O}, \text{Soot}$	$\text{C}_3\text{H}_6$ Soot	$\text{CO}_2$ $\text{CO}, \text{Soot}$	$\text{C}_3\text{H}_6$ $\text{H}_2\text{O}, \text{Soot}$	$\text{C}_3\text{H}_6, \text{H}_2\text{O}$ $\text{CO}_2$	
$\lambda$ ( $\mu\text{m}$ )	1.00	2.63	3.08	3.85	5.13	8.51	200

Table 6.5: Limits of the spectral bands for propane ( $\text{C}_3\text{H}_8$ ).

$\omega$ (1/cm)	10000	3800	3350	2550	1650	1175	50
6 Band Model	1	2	3	4	5	6	
Major Species	Soot $\text{CO}_2, \text{H}_2\text{O}$	$\text{CO}_2$ $\text{H}_2\text{O}, \text{Soot}$	$\text{C}_3\text{H}_8$ Soot	$\text{CO}_2$ $\text{CO}, \text{H}_2\text{O}, \text{Soot}$	$\text{C}_3\text{H}_8$ $\text{H}_2\text{O}, \text{Soot}$	$\text{H}_2\text{O}$ $\text{CO}_2$	
$\lambda$ ( $\mu\text{m}$ )	1.00	2.63	2.99	3.92	6.06	8.51	200

Table 6.6: Limits of the spectral bands for heptane ( $C_7H_{16}$ ).

$\omega$ (1/cm)	10000	3800	3250	2550	1775	1100	50
6 Band Model	1	2	3	4	5	6	
Major Species	Soot $CO_2, H_2O$	$CO_2$ $H_2O, Soot$	$C_7H_{16}$ Soot	$CO_2$ CO, Soot	$C_7H_{16}$ $H_2O, Soot$	$H_2O$ $CO_2$	
$\lambda$ ( $\mu m$ )	1.00	2.63	3.08	3.92	5.63	9.09	200

Table 6.7: Limits of the spectral bands for toluene ( $C_7H_8$ ).

$\omega$ (1/cm)	10000	3800	3200	2550	2050	1200	50
6 Band Model	1	2	3	4	5	6	
Major Species	Soot $CO_2, H_2O$	$CO_2$ $H_2O, Soot$	$C_7H_8$ Soot	$CO_2$ CO, Soot	$C_7H_8$ $H_2O, Soot$	$C_7H_8, H_2O$ $CO_2$	
$\lambda$ ( $\mu m$ )	1.00	2.63	3.12	3.92	4.88	8.33	200

Table 6.8: Limits of the spectral bands for methanol ( $CH_3OH$ ).

$\omega$ (1/cm)	10000	3825	3200	2600	1675	1125	50
6 Band Model	1	2	3	4	5	6	
Major Species	Soot $CO_2, H_2O$	$CH_3OH$ $CO_2, Soot$	$CH_3OH$ Soot	$CO_2$ CO, Soot	$CH_3OH$ $H_2O, Soot$	$CH_3OH, H_2O$ $CO_2$	
$\lambda$ ( $\mu m$ )	1.00	2.61	3.12	3.85	5.97	8.89	200

Table 6.9: Limits of the spectral bands for methyl methacrylate (MMA).

$\omega$ (1/cm)	10000	3800	3250	2650	2050	1250	50
6 Band Model	1	2	3	4	5	6	
Major Species	Soot $CO_2, H_2O$	$CO_2$ $H_2O, Soot$	MMA Soot	$CO_2$ CO, Soot	MMA $H_2O, Soot$	MMA, $H_2O$ $CO_2$	
$\lambda$ ( $\mu m$ )	1.00	2.63	3.08	3.77	4.88	8.00	200

### 6.1.3 Correction of the Emission Source Term

In calculations of limited spatial resolution, the source term,  $I_b$ , defined in Eq. (6.6) requires special treatment in the flaming region of the fire. Typical FDS calculations use grid cells that are tens of centimeters in size, and consequently the computed temperatures constitute a bulk average for a given grid cell and are considerably lower than the maximum temperature in a diffusion flame. Because of its fourth-power dependence on the temperature, the source term must be modeled in those grid cells where combustion occurs. Elsewhere, the computed temperature is used directly to compute the source term. It is assumed that this “flaming region” is where the local, nominal radiative loss is greater than a specified lower bound,  $\chi_r q''' > 10 \text{ kW/m}^3$ . In this region, the global radiative fraction model is used. The emission source term is multiplied by a corrective factor,  $C$ :

$$I_{b,f}(\mathbf{x}) = C \frac{\sigma T(\mathbf{x})^4}{\pi} ; \quad C = \min \left( 100, \max \left[ 1, \frac{\sum_{\chi_r q'''_{ijk} > 10} (\chi_r q'''_{ijk} + \kappa_{ijk} U_{ijk}) V_{ijk}}{\sum_{\chi_r q'''_{ijk} > 10} (4 \kappa_{ijk} \sigma T_{ijk}^4) V_{ijk}} \right] \right) \quad (6.10)$$

When the source term defined in Eq. (6.10) is substituted into Eq. (6.8), the net radiative emission from the flaming region becomes the desired fraction of the total heat release rate. Note that this correction factor is bounded below by 1 and above by 100. The upper bound is arbitrary, meant to prevent spurious behavior at the start of a simulation.

The radiative fraction,  $\chi_r$ , is a useful quantity in fire science. It is the nominal fraction of the combustion energy that is emitted as thermal radiation. For most combustibles,  $\chi_r$  is between 0.3 and 0.4 [60]. However, in Eq. (6.10),  $\chi_r$  is interpreted as the fraction of energy radiated from the flaming region. For a small fire with a base diameter less than approximately 1 m, the local  $\chi_r$  is approximately equal to its global counterpart. However, as the fire increases in size, the global value will typically decrease due to a net re-absorption of the thermal radiation by the increasing smoke mantle [61].

To account for the possibility that different fuels may have different values of radiative fraction,  $\chi_r$  is defined on a per-reaction basis, similar to the approach discussed by Gupta et al. [62]. If multiple  $\chi_r$  values are given, FDS will generate a local  $\chi_r$  by weighting the reaction specific  $\chi_r$  values by the local reaction rates.

## 6.2 Numerical Method

This section describes how  $\nabla \cdot \dot{\mathbf{q}}_r''$  (the radiative loss term) is computed at all gas-phase cells, and how the radiative heat flux  $\dot{q}_r''$  is computed at solid boundaries.

### 6.2.1 Angular Discretization

To obtain the discretized form of the RTE, the unit sphere is divided into a finite number of solid angles. The coordinate system used to discretize the solid angle is shown in Figure 6.1. The discretization of the solid angle is done by dividing first the polar angle,  $\theta$ , into  $N_\theta$  bands, where  $N_\theta$  is an even integer. Each  $\theta$ -band is then divided into  $N_\phi(\theta)$  parts in the azimuthal ( $\phi$ ) direction.  $N_\phi(\theta)$  must be divisible by 4. The numbers  $N_\theta$  and  $N_\phi(\theta)$  are chosen to give the total number of angles  $N_\Omega$  as close to the value defined by the user as possible.  $N_\Omega$  is calculated as

$$N_\Omega = \sum_{i=1}^{N_\theta} N_\phi(\theta_i) \quad (6.11)$$

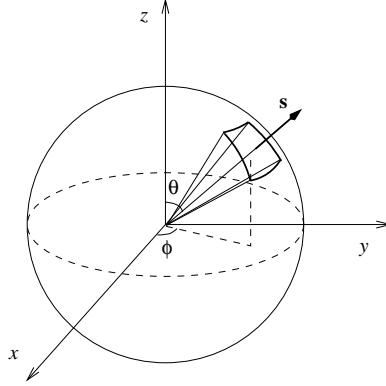


Figure 6.1: Coordinate system of the angular discretization.

The distribution of the angles is based on empirical rules that try to produce equal solid angles  $\delta\Omega^l = 4\pi/N_\Omega$ . The number of  $\theta$ -bands is

$$N_\theta = 1.17 N_\Omega^{1/2.26} \quad (6.12)$$

rounded to the nearest even integer. The number of  $\phi$ -angles on each band is

$$N_\phi(\theta) = \max \{ 4, 0.5 N_\Omega [\cos(\theta^-) - \cos(\theta^+)] \} \quad (6.13)$$

rounded to the nearest integer that is divisible by 4.  $\theta^-$  and  $\theta^+$  are the lower and upper bounds of the  $\theta$ -band, respectively. The discretization is symmetric with respect to the planes  $x = 0$ ,  $y = 0$ , and  $z = 0$ . This symmetry has three important benefits: First, it avoids the problems caused by the fact that the first-order upwind scheme, used to calculate intensities on the cell boundaries, is more diffusive in non-axial directions than axial. Second, the treatment of the mirror boundaries becomes very simple, as will be shown later. Third, it avoids so called “overhang” situations, where  $\mathbf{s} \cdot \mathbf{i}$ ,  $\mathbf{s} \cdot \mathbf{j}$  or  $\mathbf{s} \cdot \mathbf{k}$  changes sign inside the control angle. These “overhangs” would make the resulting system of linear equations more complicated.

In the axially symmetric case these “overhangs” cannot be avoided, and a special treatment, developed by Murthy and Mathur [63], is applied. In these cases  $N_\phi(\theta_i)$  is kept constant, and the total number of angles is  $N_\Omega = N_\theta \times N_\phi$ . In addition, the angle of the vertical slice of the cylinder is chosen to be the same as  $\delta\phi$ .

## 6.2.2 Spatial Discretization

The grid used for the RTE solver is the same as for the hydrodynamic solver. The radiative transport equation (6.2) is solved using techniques similar to those for convective transport in finite volume methods for fluid flow [64]; thus, the name given to it is the Finite Volume Method (FVM). More details of the model implementation are included in Ref. [65].

The thermal radiation spectrum is first divided into bands, as described in section 6.1. The procedure outlined below is applied for each band of a wide band model, and thus the subscript  $n$  has been removed for clarity. The unit sphere is then discretized, as described in section 6.2.1. Finally, the computational domain is divided into numerical grid as described in section 2.2. In each grid cell, a discretized equation is derived by integrating Eq. (6.2) over the volume of cell  $ijk$  and the control angle  $\delta\Omega^l$ , to obtain

$$\int_{\delta\Omega^l} \int_{V_{ijk}} \mathbf{s} \cdot \nabla I(\mathbf{x}, \mathbf{s}) dV d\Omega = \int_{\delta\Omega^l} \int_{V_{ijk}} \kappa(\mathbf{x}) [I_b(\mathbf{x}) - I(\mathbf{x}, \mathbf{s})] dV d\Omega \quad (6.14)$$

The volume integral on the left hand side is replaced by a surface integral over the cell faces using the divergence theorem.

$$\int_{\delta\Omega^l} \int_{A_{ijk}} (\mathbf{s} \cdot \mathbf{n}) I(\mathbf{x}, \mathbf{s}) dA d\Omega = \int_{\delta\Omega^l} \int_{V_{ijk}} \kappa(\mathbf{x}) [I_b(\mathbf{x}) - I(\mathbf{x}, \mathbf{s})] dV d\Omega \quad (6.15)$$

Assuming that the radiation intensity  $I(\mathbf{x}, \mathbf{s})$  is constant on each of the cell faces, the surface integral can be approximated by a sum over the cell faces. Assuming further that  $I(\mathbf{x}, \mathbf{s})$  is constant within the volume  $V_{ijk}$  and over the angle  $\delta\Omega^l$ , and that  $\kappa(\mathbf{x})$  and  $I_b(\mathbf{x})$  are constants within the volume  $V_{ijk}$ , we obtain

$$\sum_{m=1}^6 A_m I_m^l \int_{\Omega^l} (\mathbf{s} \cdot \mathbf{n}_m) d\Omega = \kappa_{ijk} [I_{b,ijk} - I_{ijk}^l] V_{ijk} \delta\Omega^l \quad (6.16)$$

where

$I_{ijk}^l$	radiant intensity in direction $l$
$I_m^l$	radiant intensity at cell face $m$
$I_{b,ijk}$	radiant blackbody intensity in cell
$\delta\Omega^l$	solid angle corresponding to direction $l$
$V_{ijk}$	volume of cell $ijk$
$A_m$	area of cell face $m$
$\mathbf{n}_m$	unit normal vector of the cell face $m$

Note that while the intensity is assumed constant within the angle  $\delta\Omega^l$ , its direction covers the angle  $\delta\Omega^l$  exactly. The local incident radiation intensity is

$$U_{ijk} = \sum_{l=1}^{N_\Omega} I_{ijk}^l \delta\Omega^l \quad (6.17)$$

In Cartesian coordinates<sup>1</sup>, the normal vectors  $\mathbf{n}_m$  are the base vectors of the coordinate system and the integrals over the solid angle do not depend on the physical coordinate, but the direction only. These integrals are denoted as

$$D_m^l = \int_{\Omega^l} (\mathbf{s} \cdot \mathbf{n}_m) d\Omega \quad (6.18)$$

and the discrete equation becomes

$$\sum_{m=1}^6 A_m I_m^l D_m^l = \kappa_{ijk} [I_{b,ijk} - I_{ijk}^l] V_{ijk} \delta\Omega^l \quad (6.19)$$

The cell face intensities,  $I_m^l$  appearing on the left hand side of (6.19) are calculated using a first order upwind scheme. Consider, for example, a control angle having a direction vector  $\mathbf{s}$ . If the radiation is traveling in the positive  $x$ -direction, i.e.,  $\mathbf{s} \cdot \mathbf{i} \geq 0$ , the intensity on the upwind side,  $I_{xu}^l$  is assumed to be the intensity in the neighboring cell,  $I_{i-1,jk}^l$ , and the intensity on the downwind side is the (unknown) intensity in the cell itself  $I_{ijk}^l$ . The discrete RTE can now be written using the upwind intensities  $I_{xu}^l$ ,  $I_{yu}^l$  and  $I_{zu}^l$  and  $I_{ijk}^l$ :

$$\begin{aligned} & A_x I_{xu}^l D_{xu}^l + A_x I_{ijk}^l D_{xd}^l + \\ & A_y I_{yu}^l D_{yu}^l + A_y I_{ijk}^l D_{yd}^l + \end{aligned}$$

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<sup>1</sup>In the axisymmetric case equation (6.16) becomes a little bit more complicated, as the cell face normal vectors  $\mathbf{n}_m$  are not always constant. However, the computational efficiency can still be retained.

$$A_z I_{zu}^l D_{zu}^l + A_z I_{ijk}^l D_{zd}^l = \kappa_{ijk} I_{b,ijk} V_{ijk} \delta\Omega^l - \kappa_{ijk} I_{ijk}^l V_{ijk} \delta\Omega^l \quad (6.20)$$

where the  $D$ -terms on the LHS are integrals (6.18) evaluated on upwind and downwind sides of the cell. In the rectilinear mesh,  $D_{xu}^l = -D_{xd}^l$  and the equation can be simplified further. In addition, the integrals over the solid angle can be calculated analytically

$$\begin{aligned} D_x^l &= \int_{\Omega'} (\mathbf{s}^l \cdot \mathbf{i}) d\Omega = \int_{\delta\phi} \int_{\delta\theta} (\mathbf{s}^l \cdot \mathbf{i}) \sin\theta d\theta d\phi = \int_{\delta\phi} \int_{\delta\theta} \cos\phi \sin\theta \sin\theta d\theta d\phi \\ &= \frac{1}{2} (\sin\phi^+ - \sin\phi^-) [\Delta\theta - (\cos\theta^+ \sin\theta^+ - \cos\theta^- \sin\theta^-)] \end{aligned} \quad (6.21)$$

$$\begin{aligned} D_y^l &= \int_{\Omega'} (\mathbf{s}^l \cdot \mathbf{j}) d\Omega = \int_{\delta\phi} \int_{\delta\theta} \sin\phi \sin\theta \sin\theta d\theta d\phi \\ &= \frac{1}{2} (\cos\phi^- - \cos\phi^+) [\Delta\theta - (\cos\theta^+ \sin\theta^+ - \cos\theta^- \sin\theta^-)] \end{aligned} \quad (6.22)$$

$$\begin{aligned} D_z^l &= \int_{\Omega'} (\mathbf{s}^l \cdot \mathbf{k}) d\Omega = \int_{\delta\phi} \int_{\delta\theta} \cos\theta \sin\theta d\theta d\phi \\ &= \frac{1}{2} \Delta\phi [(\sin\theta^+)^2 - (\sin\theta^-)^2] \end{aligned} \quad (6.23)$$

$$\delta\Omega^l = \int_{\Omega'} d\Omega = \int_{\delta\phi} \int_{\delta\theta} \sin\theta d\theta d\phi \quad (6.24)$$

Equation (6.20) for the unknown intensity  $I_{ijk}^l$  is written in the form

$$a_{ijk}^l I_{ijk}^l = a_x^l I_{xu}^l + a_y^l I_{yu}^l + a_z^l I_{zu}^l + b_{ijk}^l \quad (6.25)$$

where

$$a_{ijk}^l = A_x |D_x^l| + A_y |D_y^l| + A_z |D_z^l| + \kappa_{ijk} V_{ijk} \delta\Omega^l \quad (6.26)$$

$$a_x^l = A_x |D_x^l| \quad (6.27)$$

$$a_y^l = A_y |D_y^l| \quad (6.28)$$

$$a_z^l = A_z |D_z^l| \quad (6.29)$$

$$b_{ijk}^l = \kappa_{ijk} I_{b,ijk} V_{ijk} \delta\Omega^l \quad (6.30)$$

Here  $\mathbf{i}$ ,  $\mathbf{j}$  and  $\mathbf{k}$  are the base vectors of the Cartesian coordinate system.  $\theta^+$ ,  $\theta^-$ ,  $\phi^+$  and  $\phi^-$  are the upper and lower boundaries of the control angle in the polar and azimuthal directions, respectively, and  $\Delta\theta = \theta^+ - \theta^-$  and  $\Delta\phi = \phi^+ - \phi^-$ .

The solution method of (6.20) is based on an explicit marching sequence [66]. The marching direction depends on the propagation direction of the radiation intensity. As the marching is done in the “downwind” direction, the “upwind” intensities in all three spatial directions are known, and the intensity  $I_{ijk}^l$  can be solved directly from an algebraic equation. In the first cell to be solved, all the upwind intensities are determined from solid or gas phase boundaries. In theory, iterations are needed if the reflections or scattering are important, or if the scenario is optically very thick. Currently, no iterations are made.

### 6.2.3 Spatial Discretization in Cut Cells

In the case of complex geometry where irregular cut-cells may exist (see Sec. 12) an unstructured version of the RTE in Eq. 6.19 is adopted

$$\sum_{f=1}^n A_f I_f^l D_f^l = \kappa_{ijk} \left[ I_{b,ijk} - I_{ijk}^l \right] V_{ijk} \delta\Omega^l, \quad (6.31)$$

where  $A_f$  are the  $n$  cut-face areas. These include both gasphase cut-faces, which are cartesian grid aligned, and in-boundary cut-faces, which can have any orientation. The  $D_f$ -terms are calculated as

$$\begin{aligned} D_f^l &= \int_{\Omega^l} (\mathbf{s} \cdot \mathbf{n}_f) d\Omega \\ &= \int_{\delta\phi} \int_{\delta\theta} [\cos\phi \sin\theta \hat{x}_f + \sin\phi \sin\theta \hat{y}_f + \cos\theta \hat{z}_f] \sin\theta d\theta d\phi \\ &= D_x^l \hat{x}_f + D_y^l \hat{y}_f + D_z^l \hat{z}_f, \end{aligned} \quad (6.32)$$

with  $D_x^l$ ,  $D_y^l$ , and  $D_z^l$  defined as above and  $(\hat{x}_f, \hat{y}_f, \hat{z}_f)$  is the unit normal vector of the relevant face. If  $D_f^l > 0$  the faces are upwind for a given angular direction,  $D_{fu}^l$ , and downwind otherwise,  $D_{fd}^l$ . Equation 6.20 can then be rewritten as

$$\sum_{fd} A_{fd} I_{ijk}^l D_{fd}^l = \kappa_{ijk} I_{b,ijk} V_{ijk} \delta\Omega^l - \kappa_{ijk} I_{ijk}^l V_{ijk} \delta\Omega^l. \quad (6.33)$$

In this case the upwind cut-face intensities,  $I_{nu}^l$ , come from the upwind value for gasphase cut-faces (as with the cartesian solution) and from the boundary conditions for in-boundary cut-faces. The downwind cut-face intensities,  $I_{nd}^l$ , use the cell value,  $I_{ijk}^l$ .

Adopting the previous notation, the equation for intensity becomes

$$a_{ijk}^l I_{ijk}^l = \sum_{nu} a_{nu} I_{nu}^l + b_{ijk}^l \quad (6.34)$$

where

$$a_{ijk}^l = \sum_{nd} A_{nd} [D_x^l \hat{x}_{nd} + D_y^l \hat{y}_{nd} + D_z^l \hat{z}_{nd}] + \kappa_{ijk} V_{ijk} \delta\Omega^l \quad (6.35)$$

$$a_{nu}^l = A_{nu} [D_x^l \hat{x}_{nu} + D_y^l \hat{y}_{nu} + D_z^l \hat{z}_{nu}] \quad (6.36)$$

$$b_{ijk}^l = \kappa_{ijk} I_{b,ijk} V_{ijk} \delta\Omega^l \quad (6.37)$$

### 6.2.4 Boundary Conditions

The boundary condition for the radiation intensity leaving a gray diffuse wall is given as

$$I_w(\mathbf{s}) = \frac{\varepsilon \sigma T_w^4}{\pi} + \frac{1-\varepsilon}{\pi} \int_{\mathbf{s} \cdot \mathbf{n}_w < 0} I_w(\mathbf{s}) |\mathbf{s} \cdot \mathbf{n}_w| d\Omega \quad (6.38)$$

where  $I_w(\mathbf{s})$  is the intensity at the wall,  $\varepsilon$  is the emissivity, and  $T_w$  is the wall surface temperature. In discretized form, the boundary condition on a solid wall is given as

$$I_w^l = \frac{\varepsilon \sigma T_w^4}{\pi} + \frac{1-\varepsilon}{\pi} \sum_{D_w^{l'} < 0} I_w^{l'} |D_w^{l'}| \quad (6.39)$$

where  $D_w^{l'} = \int_{\Omega'} (\mathbf{s} \cdot \mathbf{n}_w) d\Omega$ . The constraint  $D_w^{l'} < 0$  means that only the “incoming” directions are taken into account when calculating the reflection. The *net* radiative heat flux on the wall is

$$q_r'' = \sum_{l=1}^{N_\Omega} I_w^l \int_{\delta\Omega'} (\mathbf{s} \cdot \mathbf{n}_w) d\Omega = \sum_{l=1}^{N_\Omega} I_w^l D_n^l \quad (6.40)$$

where the coefficients  $D_n^l$  are equal to  $\pm D_x^l$ ,  $\pm D_y^l$  or  $\pm D_z^l$ , and can be calculated for each wall element at the start of the calculation.

The open boundaries are treated as black walls, where the incoming intensity is the blackbody intensity of the ambient temperature. On mirror boundaries the intensities leaving the wall are calculated from the incoming intensities using a predefined connection matrix:

$$I_{w,ijk}^l = I^{l'} \quad (6.41)$$

Computationally intensive integration over all the incoming directions is avoided by keeping the solid angle discretization symmetric on the  $x$ ,  $y$  and  $z$  planes. The connection matrix associates one incoming direction  $l'$  to each mirrored direction on each wall cell.

## 6.3 Absorption and Scattering by Small Droplets and Particles

The attenuation of thermal radiation by liquid droplets and small particles is an important consideration, especially for water mist systems [67]. Droplets and particles attenuate thermal radiation through a combination of scattering and absorption [68]. The radiation-spray interaction must therefore be solved for both the accurate prediction of the radiation field and for the particle energy balance. If the gas phase absorption and emission in Eq. (6.1) are temporarily neglected for simplicity, the radiative transport equation becomes

$$\mathbf{s} \cdot \nabla I_\lambda(\mathbf{x}, \mathbf{s}) = -[\kappa_p(\mathbf{x}, \lambda) + \sigma_p(\mathbf{x}, \lambda)] I_\lambda(\mathbf{x}, \mathbf{s}) + \kappa_p(\mathbf{x}, \lambda) I_{b,p}(\mathbf{x}, \lambda) + \frac{\sigma_p(\mathbf{x}, \lambda)}{4\pi} \int_{4\pi} \Phi(\mathbf{s}', \mathbf{s}) I_\lambda(\mathbf{x}, \mathbf{s}') d\Omega \quad (6.42)$$

where  $\kappa_p$  is the particle absorption coefficient,  $\sigma_p$  is the particle scattering coefficient and  $I_{b,p}$  is the emission term of the particles.  $\Phi(\mathbf{s}', \mathbf{s})$  is a scattering phase function that gives the scattered intensity fraction from direction  $\mathbf{s}'$  to  $\mathbf{s}$ .

### 6.3.1 Absorption and Scattering Coefficients

Radiation absorption and scattering by particles depends on their cross sectional areas and radiative material properties. For simplicity, we assume that the particles are spherical in shape, in which case the cross sectional area of a particle is  $\pi r^2$ , where  $r$  is the particle radius. If the local number density distribution of particles at location  $\mathbf{x}$  is denoted by  $n(r(\mathbf{x}))$ , the local absorption and scattering coefficients within a spray/particle cloud can be calculated from:

$$\kappa_p(\mathbf{x}, \lambda) = \int_0^\infty n(r(\mathbf{x})) Q_a(r, \lambda) \pi r^2 dr \quad (6.43)$$

$$\sigma_p(\mathbf{x}, \lambda) = \int_0^\infty n(r(\mathbf{x})) Q_s(r, \lambda) \pi r^2 dr \quad (6.44)$$

where  $Q_a$  and  $Q_s$  are the absorption and scattering efficiencies, respectively. For the computation of the spray/particle cloud radiative properties, spherical particles are assumed and the radiative properties of the individual particles are computed using Mie theory.

Based on Refs. [69] and [70], the real particle size distribution inside a grid cell is modeled as a mono-disperse suspension whose diameter corresponds to the Sauter mean ( $d_{32}$ ) diameter of the poly-disperse spray. This assumption leads to a simplified expression of the radiative coefficients

$$\kappa_p(\mathbf{x}, \lambda) = A_p(\mathbf{x}) Q_a(r_{32}, \lambda) \quad (6.45)$$

$$\sigma_p(\mathbf{x}, \lambda) = A_p(\mathbf{x}) Q_s(r_{32}, \lambda) \quad (6.46)$$

These expressions are functions of the total cross sectional area per unit volume of the droplets,  $A_p$ , which is computed simply by summing the cross sectional areas of all the droplets within a cell and dividing by the cell volume. For practical reasons, a relaxation factor of 0.5 is used to smooth slightly the temporal variation of  $A_p$ .

### 6.3.2 Approximating the In-Scattering Integral

An accurate computation of the in-scattering integral on the right hand side of Eq. (6.42) would be extremely time consuming and require a prohibitive amount of memory because the individual intensities in each location would have to be stored. Instead, a simplified form of Eq. (6.42) can be derived:

$$\mathbf{s} \cdot \nabla I_\lambda(\mathbf{x}, \mathbf{s}) = -[\kappa_p(\mathbf{x}, \lambda) + \bar{\sigma}_p(\mathbf{x}, \lambda)] I_\lambda(\mathbf{x}, \mathbf{s}) + \kappa_p(\mathbf{x}, \lambda) I_{b,p}(\mathbf{x}, \lambda) + \frac{\bar{\sigma}_p(\mathbf{x}, \lambda)}{4\pi} U(\mathbf{x}, \lambda) \quad (6.47)$$

where  $U(\mathbf{x})$  is the total intensity integrated over the unit sphere and  $\bar{\sigma}_p$  is an effective scattering coefficient. The derivation of Eq. (6.47) is given in Appendix J. This simplified equation can be integrated over the spectrum to get the band specific RTE's. The procedure is exactly the same as that used for the gas phase RTE. After the band integrations, the spray RTE for band  $n$  becomes:

$$\mathbf{s} \cdot \nabla I_n(\mathbf{x}, \mathbf{s}) = - [\kappa_{p,n}(\mathbf{x}) + \bar{\sigma}_{p,n}(\mathbf{x})] I_n(\mathbf{x}, \mathbf{s}) + \kappa_{p,n}(\mathbf{x}) I_{b,p,n}(\mathbf{x}) + \frac{\bar{\sigma}_{p,n}(\mathbf{x})}{4\pi} U_n(\mathbf{x}) \quad (6.48)$$

where the source function is based on the average particle temperature within a cell.

### 6.3.3 Forward Fraction of Scattering

The effective scattering coefficient in Eq. (6.47) is defined:

$$\bar{\sigma}_p(\mathbf{x}, \lambda) = \frac{4\pi}{4\pi - \delta\Omega^l} (1 - \chi_f(\mathbf{x}, \lambda)) \sigma_p(\mathbf{x}, \lambda) \quad (6.49)$$

where  $\chi_f$  is the fraction of the total intensity originally within the solid angle  $\delta\Omega^l$  that is scattered into the same angle,  $\delta\Omega^l$ . The computation of  $\chi_f$  has been derived in Ref. [71]. It can be shown that here  $\chi_f$  becomes:

$$\chi_f(r, \lambda) = \frac{1}{\delta\Omega^l} \int_{\mu'}^1 \int_{\mu'}^1 \int_{\mu_{p,0}}^{\mu_{p,\pi}} \frac{P_0(\theta_p, \lambda)}{[(1 - \mu^2)(1 - \mu'^2) - (\mu_p - \mu\mu')^2]^{1/2}} d\mu_p d\mu d\mu' \quad (6.50)$$

where  $\mu_p$  is a cosine of the scattering angle  $\theta_p$  and  $P_0(\theta_p, \lambda)$  is a single droplet scattering phase function

$$P_0(\theta_p, \lambda) = \frac{\lambda^2 (|S_1(\theta_p)|^2 + |S_2(\theta_p)|^2)}{2Q_s(r, \lambda)\pi r^2} \quad (6.51)$$

$S_1(\theta_p)$  and  $S_2(\theta_p)$  are the scattering amplitudes, given by Mie-theory. The integration limit,  $\mu^l$ , is a cosine of the polar angle defining the boundary of the symmetric control angle,  $\delta\Omega^l$

$$\mu^l = \cos(\theta^l) = 1 - \frac{2}{N_\Omega} \quad (6.52)$$

The limits of the innermost integral are

$$\mu_{p,0} = \mu\mu' + \sqrt{1 - \mu^2}\sqrt{1 - \mu'^2} \quad ; \quad \mu_{p,\pi} = \mu\mu' - \sqrt{1 - \mu^2}\sqrt{1 - \mu'^2} \quad (6.53)$$

### 6.3.4 Solution Procedure

The absorption and scattering coefficients,  $\kappa_p$  and  $\bar{\sigma}_p$  are not repeatedly calculated during the simulation. Instead, they are tabulated at the beginning of the simulation for each band and a range of different Sauter mean diameters,  $r_{32}$ . The averaged quantities, now functions  $r_{32}$  only, are stored in one-dimensional arrays. During the simulation, the local properties are found by table look-up.

In the band integration of  $\kappa_p$  and  $\bar{\sigma}_p$ , a constant “radiation” temperature,  $T_{rad}$ , is used to provide the wavelength weighting (Planck function).  $T_{rad}$  should be selected to represent a typical radiating flame temperature. A value of 1173 K is used by default.

The absorption and scattering efficiencies,  $Q_a$  and  $Q_s$ , and the scattering phase function  $P_0(\theta_p, \lambda)$ , are calculated using the MieV code developed by Wiscombe [72]. The spectral properties of the particles can be specified in terms of a wavelength-dependent complex refractive index. Pre-compiled data are included for water and a generic hydrocarbon fuel based on diesel fuel and heptane. For water, the values of the imaginary part of the complex refractive index (related to the absorption coefficient) are taken from Ref. [73], and a

value of 1.33 is used for the real part. For fuel, the droplet spectral properties are taken from Ref. [74], which includes measurements of the refractive index of a diesel fuel and a comparison of the values with those of heptane. The diesel properties are used for the real part of the refractive index. For the complex part, the heptane properties are used to avoid the uncertainty associated with different types of diesel fuels. Usually, the radiative properties of the particle cloud are more sensitive to the particle size and concentration than to values of the refractive index.

### 6.3.5 Heat Absorbed by Droplets and Small Particles

The droplet contribution to the radiative loss term is

$$\dot{q}_r''' \equiv -\nabla \cdot \dot{\mathbf{q}}_r''(\mathbf{x}) (\text{droplets}) = \kappa_p(\mathbf{x}) [U(\mathbf{x}) - 4\pi I_{b,p}(\mathbf{x})] \quad (6.54)$$

For each individual droplet, the radiative heating/cooling power is computed as

$$\dot{q}_r = \frac{m_p}{\rho_p(\mathbf{x})} \kappa_p(\mathbf{x}) [U(\mathbf{x}) - 4\pi I_{b,p}(\mathbf{x})] \quad (6.55)$$

where  $m_p$  is the mass of the droplet and  $\rho_p(\mathbf{x})$  is the total density of droplets in the cell.

## 6.4 Absorption by Large, Non-Scattering Particles

FDS uses Lagrangian particles to represent objects that are not resolvable on the numerical grid, but are still relatively large compared to liquid droplets. For example, vegetation is modeled as a collection of large, opaque, non-scattering particles. For more discussion of the radiation absorption by large particles, see Sec. 9.6.



# Chapter 7

## Solid Phase

FDS assumes that solid surfaces consist of multiple layers, with each layer composed of multiple material components that can undergo multiple thermal degradation reactions. Heat conduction is assumed only in the direction normal to the surface. Each reaction can produce multiple gas and solid species. This chapter describes the heat conduction equation for solid materials, plus the various coefficients, source terms, and boundary conditions, including the computation of the convective heat flux  $\dot{q}_c''$  at solid boundaries.

### 7.1 The Heat Conduction Equation for a Solid

The one-dimensional heat conduction equation for the solid phase temperature  $T_s(x, t)$  is applied in the direction  $x$  pointing into the solid (the point  $x = 0$  represents the surface)<sup>1</sup>

$$\rho_s c_s \frac{\partial T_s}{\partial t} = \frac{\partial}{\partial x} \left( k_s \frac{\partial T_s}{\partial x} \right) + \dot{q}_s''' \quad (7.2)$$

Sec. 7.1.3 describes the component-averaged material properties,  $k_s$  and  $\rho_s c_s$ . The source term,  $\dot{q}_s'''$ , consists of chemical reactions and radiative absorption:

$$\dot{q}_s''' = \dot{q}_{s,c}''' + \dot{q}_{s,r}''' \quad (7.3)$$

Sec. 7.2 describes the term  $\dot{q}_{s,c}'''$ , which is essentially the heat production (loss) rate given by the pyrolysis models for different types of solid and liquid fuels. Sec. 7.1.1 describes the term  $\dot{q}_{s,r}'''$ , the radiative absorption and emission in depth.

The boundary condition on the front surface of a solid obstruction is

$$-k_s \frac{\partial T_s}{\partial x}(0, t) = \dot{q}_c'' + \dot{q}_r'' \quad (7.4)$$

where  $\dot{q}_c''$  is the convective and  $\dot{q}_r''$  the radiative flux. If the radiation is assumed to penetrate in depth, the surface radiation term,  $\dot{q}_r''$ , is set to 0. Sec. 7.1.2 describes the convective heat transfer to the solid surface.

On the back surface, there are two possible boundary conditions: (1) if the back surface is assumed to be open either to an ambient void or to another part of the computational domain, the back side boundary

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<sup>1</sup>In cylindrical and spherical coordinates, the heat conduction equation is written

$$\rho_s c_s \frac{\partial T_s}{\partial t} = \frac{1}{r} \frac{\partial}{\partial r} \left( r k_s \frac{\partial T_s}{\partial r} \right) + \dot{q}_s''' \quad ; \quad \rho_s c_s \frac{\partial T_s}{\partial t} = \frac{1}{r^2} \frac{\partial}{\partial r} \left( r^2 k_s \frac{\partial T_s}{\partial r} \right) + \dot{q}_s''' \quad (7.1)$$

FDS offers the user these options for cases where the obstruction surface is not flat, but rather cylindrical or spherical in shape. This option is useful in describing the behavior of small, complicated “targets” like cables or heat detection devices.

condition is similar to that of the front side, or (2) if the back side is assumed to be perfectly insulated, an adiabatic condition is used

$$-k_s \frac{\partial T_s}{\partial x} = 0 \quad (7.5)$$

The numerical solution of the solid phase heat equation is presented in detail in Appendix L.

### 7.1.1 Radiation Heat Transfer to Solids

If it is assumed that the thermal radiation from the surrounding gases is absorbed within an infinitely thin layer at the surface of the solid obstruction, then the net radiative heat flux is the sum of incoming and outgoing components,  $\dot{q}_r'' = \dot{q}_{r,\text{in}}'' - \dot{q}_{r,\text{out}}''$ :

$$\dot{q}_{r,\text{in}}'' = \varepsilon \int_{\mathbf{s}' \cdot \mathbf{n}_w < 0} I_w(\mathbf{s}') |\mathbf{s}' \cdot \mathbf{n}_w| d\Omega \quad (7.6)$$

$$\dot{q}_{r,\text{out}}'' = \varepsilon \sigma T_w^4 \quad (7.7)$$

However, many common materials are not opaque; thus, the radiation penetrates the material to some finite depth. The radiative transport within the solid (or liquid) can be described as a source term in Eq. (7.2). A “two-flux” model based on the Schuster-Schwarzschild approximation [57] assumes the radiative intensity is constant inside the “forward” and “backward” hemispheres. The transport equation for the intensity in the “forward” direction is

$$\frac{1}{2} \frac{dI^+(x)}{dx} = \kappa_s (I_b - I^+(x)) \quad (7.8)$$

where  $x$  is the distance from the material surface and  $\kappa_s$  is the component-averaged absorption coefficient:

$$\kappa_s = \sum_{\alpha=1}^{N_m} X_\alpha \kappa_{s,\alpha} \quad (7.9)$$

A corresponding formula can be given for the “backward” direction. Multiplying Eq. 7.8 by  $\pi$  gives us the “forward” radiative heat flux into the solid

$$\frac{1}{2} \frac{d\dot{q}_r^+(x)}{dx} = \kappa_s (\sigma T_s^4 - \dot{q}_r^+(x)) \quad (7.10)$$

The radiative source term in the heat conduction equation is the sum of the “forward” and “backward” flux gradients

$$\dot{q}_{s,r}'''(x) = \frac{d\dot{q}_r^+(x)}{dx} + \frac{d\dot{q}_r^-(x)}{dx} \quad (7.11)$$

The boundary condition for Eq. 7.10 at the solid (or liquid) surface is given by

$$\dot{q}_r^+(0) = \dot{q}_{r,\text{in}}'' + (1 - \varepsilon) \dot{q}_r^-(0) \quad (7.12)$$

where  $\dot{q}_r^-(0)$  is the “backward” radiative heat flux at the surface. In this formulation, the surface emissivity and the internal absorption are assumed constant.

The two-flux model has not been adapted for cylindrical or spherical geometry.

### 7.1.2 Convective Heat Transfer to Solids

The calculation of the convective heat flux depends on whether one is performing a direct numerical simulation (DNS) or a large eddy simulation (LES). For DNS, the convective heat transfer is calculated directly from the resolved gas and solid phase variables. For LES, there are a variety of empirical options.

## Direct Numerical Simulation

In a DNS calculation, the convective heat flux to a solid surface  $\dot{q}_c''$  is obtained directly from the gas temperature gradient at the boundary

$$\dot{q}_c'' = -k \frac{\partial T}{\partial n} = -k \frac{T_w - T_g}{\delta n/2} \quad (7.13)$$

where  $k$  is the thermal conductivity of the gas,  $n$  is the spatial coordinate pointing into the solid,  $\delta n$  is the normal grid spacing,  $T_g$  is the gas temperature in the center of the first gas phase cell, and  $T_w$  is the wall surface temperature.

## Empirical Natural/Forced Convection Model

In an LES calculation, the convective heat transfer coefficient,  $h$ , is taken as the maximum of its free (natural) and forced forms:

$$\dot{q}_c'' = h(T_g - T_w) ; \quad h = \frac{k}{L} \max \left( \text{Nu}_{\text{free}}, \text{Nu}_{\text{forced}}, \frac{2L}{\delta n} \right) \quad (7.14)$$

where  $T_g$  is the gas temperature in the gas phase cell adjacent to the surface,  $T_w$  is the wall (surface) temperature,  $L$  is a characteristic length, and  $k$  is the thermal conductivity of the gas. For planar surfaces,  $L$  is taken as 1 m and for spheres and cylinders,  $L$  is taken as the diameter,  $D$ . The last argument on the right-hand side corresponds to the limit for DNS cases. For Lagrangian particles the assumption is that  $D \ll \delta n$  and so the equation for  $h$  becomes:

$$h = \frac{k}{L} \max (\text{Nu}_{\text{free}}, \text{Nu}_{\text{forced}}). \quad (7.15)$$

For free (natural) convection, the Nusselt number is a function of the Rayleigh number:

$$\text{Ra} = \frac{2g|T_g - T_w|L^3}{(T_g + T_w)\nu\alpha} ; \quad \nu = \frac{\mu}{\rho} ; \quad \alpha = \frac{k}{\rho c_p} ; \quad \text{Pr} = \frac{\nu}{\alpha} \quad (7.16)$$

The following expressions are simplifications of those given in Ref. [75] under the assumption that  $\text{Pr} = 0.7$ .

$$\text{Nu}_{\text{free}} = \begin{cases} (0.825 + 0.324 \text{Ra}^{1/6})^2 & \text{Vertical plate or cylinder} \\ 0.54 \text{Ra}^{1/4} & \text{Horizontal hot plate facing up or cold plate facing down, } \text{Ra} \leq 10^7 \\ 0.15 \text{Ra}^{1/3} & \text{Horizontal hot plate facing up or cold plate facing down, } \text{Ra} > 10^7 \\ 0.52 \text{Ra}^{1/5} & \text{Horizontal hot plate facing down or cold plate facing up} \\ (0.60 + 0.321 \text{Ra}^{1/6})^2 & \text{Horizontal cylinder} \\ 2 + 0.454 \text{Ra}^{1/4} & \text{Sphere} \end{cases} \quad (7.17)$$

For forced convection, the Nusselt number takes the form:

$$\text{Nu}_{\text{forced}} = C_0 + (C_1 \text{Re}^n - C_2) \text{Pr}^m ; \quad \text{Re} = \frac{\rho |\mathbf{u}| L}{\mu} ; \quad m = 1/3 \quad (7.18)$$

The values of the coefficients are given in Table 7.1.

The impinging jet heat transfer model uses the stagnation pressure to define the velocity scale,  $U_{\text{imp}} = \sqrt{2H}$ , in the Reynolds number, but otherwise uses the forced convection Nusselt correlation with slightly modified default constants,  $C_1 = 0.55$  and  $n = 0.8$ . As with the forced convection model, custom coefficients may be used, as discussed in the FDS User Guide [3]. If the impinging jet model is implemented by the user,  $\text{Nu}_{\text{impinge}}$  gets added to the list of max arguments in Eq. (7.14).

$$\text{Nu}_{\text{impinge}} = C_0 + (C_1 \text{Re}_{\text{imp}}^n - C_2) \text{Pr}^m ; \quad \text{Re}_{\text{imp}} = \frac{\rho U_{\text{imp}} L}{\mu} ; \quad m = 1/3 \quad (7.19)$$

Table 7.1: Coefficients used for forced convection heat transfer correlations [75]

Geometry	$C_0$	$C_1$	$C_2$	$n$	Re
Flat Plate	0	0.0296	0	0.8	$\leq 10^8$
Cylinder	0	0.989	0	0.330	0.4 – 4
Cylinder	0	0.911	0	0.385	4 – 40
Cylinder	0	0.683	0	0.466	40 – 4000
Cylinder	0	0.193	0	0.618	4000 – 40000
Cylinder	0	0.027	0	0.805	40000 – 400000
Sphere	2	0.6	0	0.5	

### Optional Near-Wall Model

This section describes an optional model for the heat transfer coefficient which may be more appropriate for well-resolved LES calculations. This model has been validated for low Reynolds number heated channel flow [76] and has been used in a model to predict upper layer temperature in airplane cargo compartments [77].

Wall models aim to mimic the sudden change from molecular to turbulent transport close to the walls using algebraic formulations without resolving the smallest length scales. The theory follows dimensional analysis based on the idea that shear at the wall is constant. Accordingly, non-dimensional velocity can be defined as a function of non-dimensional length scale. In FDS, the wall model for velocity is implemented based on the law of the wall with a semi-log fit connecting the limits of the viscous and log regions (see Sec. 4.4).

By analogy to the near-wall model for velocity, the non-dimensional temperature is defined as

$$T^+ = \frac{T_g - T_w}{T_\tau} \quad (7.20)$$

where  $T_g$  is the first off-wall gas-phase cell temperature. The model profile is given by

$$T^+ = \text{Pr} y^+ \quad \text{for } y^+ \leq 11.81 \quad (7.21)$$

$$T^+ = \frac{\text{Pr}_t}{\kappa} \ln y^+ + B_T \quad \text{for } y^+ \geq 11.81 \quad (7.22)$$

where  $\text{Pr}$  and  $\text{Pr}_t$  are the molecular and turbulent Prandtl numbers ( $\text{Pr}_t = 0.5$  by default in FDS), and  $\kappa = 0.41$  is the von Kármán constant. The temperature scale,  $T_\tau$ , is defined by

$$T_\tau \equiv \frac{\dot{q}_c''}{\rho c_p u_\tau} \quad (7.23)$$

where  $\dot{q}_c''$ ,  $\rho$ ,  $c_p$  and  $u_\tau$  are the convective heat flux at the wall, the gas density, the specific heat, and the friction velocity, respectively.

The second term,  $B_T$ , on the right hand side of Eq. 7.22 is a function of the molecular Prandtl number and can be determined experimentally. Mathematically, this term is the integration constant stemming from the relation between velocity and temperature gradients. Physically, it represents the resistance to the heat and momentum transport close to the wall. FDS uses the experimental correlation proposed by Kader [78]

$$B_T = (3.85 \text{Pr}^{1/3} - 1.3)^2 + 2.12 \ln \text{Pr} \quad (7.24)$$

The convective heat transfer coefficient ( $h$ ) is obtained from the definition of  $h$  and  $T^+$ :

$$h = \frac{\dot{q}_c''}{(T_g - T_w)} = \frac{\rho c_p u_\tau}{T^+} \quad (7.25)$$

### 7.1.3 Component-Averaged Thermal Properties

The conductivity and volumetric heat capacity of the solid are defined as

$$k_s = \sum_{\alpha=1}^{N_m} X_\alpha k_{s,\alpha} ; \quad \rho_s c_s = \sum_{\alpha=1}^{N_m} \rho_{s,\alpha} c_{s,\alpha} \quad (7.26)$$

where  $N_m$  is the number of material components forming the solid,  $X_\alpha$  is the volume fraction of component  $\alpha$ , and  $\rho_{s,\alpha}$  is the *component density*:

$$\rho_{s,\alpha} = \rho_s Y_\alpha \quad (7.27)$$

where  $\rho_s$  is the density of the composite material and  $Y_\alpha$  is the mass fraction of component  $\alpha$ . The solid density is the sum of the component densities

$$\rho_s = \sum_{\alpha=1}^{N_m} \rho_{s,\alpha} \quad (7.28)$$

and the volume fraction of component  $\alpha$  is computed as

$$X_\alpha = \frac{\rho_{s,\alpha}}{\rho_\alpha} \left/ \sum_{\beta=1}^{N_m} \frac{\rho_{s,\beta}}{\rho_\beta} \right. \quad (7.29)$$

where  $\rho_\alpha$  is the true density of material  $\alpha$ . Multi-component solids are defined by specifying the mass fractions,  $Y_\alpha$ , and densities,  $\rho_\alpha$ , of the individual components of the composite.

## 7.2 Pyrolysis Models

This section describes how solid phase reactions and the chemical source term in the solid phase heat conduction equation,  $\dot{q}_{s,c}'''$  (see Eq. (7.3)), are modeled. This is commonly referred to as the “pyrolysis model”, but it actually can represent any number of reactive processes, including evaporation, charring, and internal heating. The process (enforcing consistency in material heats of formation and temperature dependent heats of reaction) via which FDS ensures energy conservation when solid phase reactions are specified, is detailed in Appendix O.

### 7.2.1 Specified Heat Release Rate

 Often the intent of a fire simulation is merely to predict the transport of smoke and heat from a *specified* fire. In other words, the heat release rate is a specified input, not something the model predicts. In these instances, the desired heat release rate is translated into a mass flux for fuel at a given solid surface, which can be thought of as the surface of a burner:

$$\dot{m}_f'' = \frac{f(t) \dot{q}_{\text{user}}''}{\Delta H_c} \quad (7.30)$$

Usually, the user specifies a desired heat release rate per unit area (HRRPUA),  $\dot{q}_{\text{user}}''$ , plus a time ramp,  $f(t)$ , and the gas phase heat of combustion,  $\Delta H_c$ . The mass flux of fuel from the surface to the gas,  $\dot{m}_f''$ , can then be computed. A special subset of this approach is where the user also specifies a heat of combustion, density, and thickness for the burning material. If the default simple chemistry model described in Sec. 5.1.2, with its single combustion reaction, is being used and there are multiple materials burning, one or more materials might have a different heat of combustion,  $\Delta H_{c,\text{solid}}$ , than the gas-phase combustion reaction. To account for this, FDS adjusts  $\dot{m}_f''$  so that the solid phase has the appropriate mass loss rate,  $\dot{m}_{f,\text{solid}}''$ .

$$\dot{m}_{f,\text{solid}}'' = \dot{m}_f'' \frac{\Delta H_c}{\Delta H_{c,\text{solid}}} \quad (7.31)$$

### 7.2.2 Scaling the Burning Rate by the Heat Flux

This approach, called Spyro, is a modified version of the specified heat release rate discussed in the prior subsection. Spyro uses data from one or more cone calorimeter or similar experiments where a sample is exposed to fixed incident radiative flux while mass loss and/or the heat release rate is measured. Data can come from either inert gasification tests or tests where pyrolyzed material is allowed to burn. A detailed justification of the approach can be found in Appendix P.

Spyro assumes that the general shape of the mass loss rate or heat release rate curve for a material sample is constant and that the curve’s magnitude and duration is a function of the incident heat flux to the sample and the sample thickness. With this assumption the burning rate for a cone test at one exposure and tested thickness can be scaled to a different exposure and thickness, i.e., the as modeled sample thickness and the incident heat flux predicted during an FDS simulation. The algorithm operates as follows:

1. Each set,  $i$ , of supplied test data requires the cone flux,  $\dot{q}_{\text{cone},i}''$ , the thickness tested in the cone  $\Delta x_i$ , and a time ramp,  $f_i(t)$ , that is applied to  $\dot{q}_{\text{user}}''$ . Note that each  $f_i(t)$  should start at  $t = 0$  corresponding to ignition since the time up to ignition is handled by the 1-D heat transfer model. For each set of data:
  - (a) Determine the thickness scaling factor  $s_t$  for data set  $i$  as  $s_{t,i} = \frac{\Delta x_{\text{ref}}}{\Delta x_i}$  where  $\Delta x_{\text{ref}}$  is the material thickness being modeled in FDS and  $\Delta x_i$  is the as tested material thickness.

- (b) Determine the incident flux over time,  $\dot{q}_{ref,i}''(t)$ . For an inert gasification test this is simply the imposed radiative flux:

$$\dot{q}_{ref,i}''(t) = \dot{q}_{cone_i}'' \quad (7.32)$$

For a cone calorimeter test, this is the imposed radiative flux minus that portion of the imposed flux,  $\Gamma_i(t)$ , that is absorbed by the flame plus heat feedback from the flame:

$$\dot{q}_{ref,i}''(t) = \dot{q}_{cone_i}''(1 - \Gamma_i(t)) + \dot{q}_{flame_i}''(t) \quad (7.33)$$

This value of  $\dot{q}_{flame_i}''(t)$  is determined using an empirical approach discussed in Appendix P.

- (c) Integrate the incident flux over time to create a curve of the energy delivered to the surface as a function of time and then invert the curve to give time vs. the energy delivered to the surface,  $t_i E_i''$ .

2. For each wall cell using the Spyro method where the ignition temperature has been reached:

- (a) For each thickness provided in the input data, integrate the FDS predicted incident flux,  $\dot{q}_{inc}''$ , using the scaled time step  $\Delta t_{scaled} = s_{t,i} \Delta t$ .
- (b) For each supplied  $f_i(t)$ , use the corresponding thickness scaled incident flux to get the equivalent time for that test data, i.e., compute  $t_i E_i''$ .
- (c) For each supplied  $f_i(t)$ , look up  $\dot{q}_{ref,i}''(t_i)$ .
- (d) For each supplied  $f_i(t)$ , determine  $\dot{q}_{user,f_i(t_i)}''$ . If  $E_i$  exceeds the range of  $f_i(t)$ , then scale the last provided data point in  $f_i(t)$  as  $f_i(t) \frac{\dot{q}_{inc}''}{\dot{q}_{ref,i}''(t_i)}$  to obtain  $\dot{q}_{\Delta x_j}''$ .
- (e) Interpolate the  $\dot{q}_{\Delta x_j}''$  to  $\Delta x_{ref}$  using  $\Delta x_j$  to obtain the HRRPUA for applied for the grid cell.

### 7.2.3 Solid Fuels

Solids can undergo simultaneous reactions under the following assumptions:

- instantaneous release of gas species
- local thermal equilibrium between the solid and gaseous components
- no condensation of gaseous products
- no porosity effects<sup>2</sup>

Each material component may undergo several competing reactions, and each of these reactions may produce some other solid component (residue) and gaseous species according to specified yield coefficients. These coefficients should sum to 1, but yields summing to less than 1 can account for products that are not explicitly included in the simulation.

The mass per unit volume of material component  $\alpha$ ,  $\rho_{s,\alpha}(x,t)$ , evolves in time according to the solid phase species conservation equation

$$\frac{\partial \rho_{s,\alpha}}{\partial t} = - \sum_{\beta=1}^{N_{r,\alpha}} r_{\alpha\beta} + S_{\alpha} \quad (7.34)$$

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<sup>2</sup>Although porosity is not explicitly included in the model, it is possible to account for it because the volume fractions defined by Eq. (7.29) need not sum to unity, in which case the thermal conductivity and absorption coefficient are effectively reduced.

where  $N_{r,\alpha}$  is the number of reactions for material  $\alpha$ ,  $r_{\alpha\beta}$  is the rate of reaction  $\beta$  in units of 1/s, and  $S_\alpha$  is the production rate of material component  $\alpha$  as a result of the reactions of the other components. The reaction rates are functions of solid and gas phase conditions and calculated as a combination of Arrhenius and power functions:

$$r_{\alpha\beta} = \underbrace{\rho_{s,\alpha}^{n_{s,\alpha\beta}}}_{\text{Reactant dependency}} \underbrace{A_{\alpha\beta} \exp\left(-\frac{E_{\alpha\beta}}{RT_s}\right)}_{\text{Arrhenius function}} \underbrace{[X_{O_2}(x)]^{n_{O_2,\alpha\beta}}}_{\text{Oxidation function}} \underbrace{T_s^{n_{t,\alpha\beta}}}_{\text{Power function}} \quad (7.35)$$

The first term describes the dependence of the reaction rate on the concentration of the reactant itself, with  $n_{s,\alpha\beta}$  being the partial reaction order. The second term is the Arrhenius function which is commonly used to describe the reaction kinetics, i.e. the dependence of the reaction rate on the material temperature. The chapter on pyrolysis in the FDS Verification Guide describes methods for determining the kinetic parameters  $A_{\alpha\beta}$  and  $E_{\alpha\beta}$  using bench-scale measurement techniques.

The third term can be used to describe the dependence on the local oxygen concentration  $X_{O_2}(x)$  and the heterogeneous reaction order,  $n_{O_2,\alpha\beta}$ . The oxygen concentration profile within practical materials depends on the competition between diffusion and reactive consumption. As FDS does not solve for the transport of gaseous species within condensed phase materials, a simple exponential profile is assumed and the user is expected to specify the characteristic depth at which oxygen would be present. The local oxygen volume fraction at depth  $x$  is calculated from the gas phase (first grid cell) oxygen volume fraction  $X_{O_2,g}$  as

$$X_{O_2}(x) = X_{O_2,g} \exp(-x/L_{g,\alpha\beta}) \quad (7.36)$$

where  $L_{g,\alpha\beta}$  is the characteristic depth of oxygen diffusion. Specifying  $L_{g,\alpha\beta} = 0$  means that the reaction takes place only at the surface of the material.

The fourth term is a power function for temperature.

The production term  $S_\alpha$  is the sum over all the reactions where the solid residue is material  $\alpha$

$$S_\alpha = \sum_{\alpha'=1}^{N_m} \sum_{\beta=1}^{N_{r,\alpha'}} v_{\alpha,\alpha'\beta} r_{\alpha'\beta} \quad (\text{where Residue}_{\alpha'\beta} = \text{Material}_\alpha) \quad (7.37)$$

where  $v_{\alpha,\alpha'\beta}$  is the yield of component  $\alpha$  from reaction  $\beta$  of component  $\alpha'$ . The volumetric production rate of each gas species,  $\gamma$ , is

$$\dot{m}_\gamma''' = \sum_{\alpha=1}^{N_m} \sum_{\beta=1}^{N_{r,\alpha}} v_{\gamma,\alpha\beta} r_{\alpha\beta} \quad (7.38)$$

It is assumed that the gases are transported instantaneously to the surface, where the mass fluxes are given by<sup>3</sup>

$$\dot{m}_\gamma'' = \int_0^L \dot{m}_\gamma'''(x) dx \quad (7.40)$$

where  $L$  is the thickness of the solid. The chemical source term in the heat conduction equation is

$$\dot{q}_{s,c}'''(x) = - \sum_{\alpha=1}^{N_m} \sum_{\beta=1}^{N_{r,\alpha}} r_{\alpha\beta}(x) H_{r,\alpha\beta} \quad (7.41)$$

where  $H_{r,\alpha\beta}$  is the heat of reaction.

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<sup>3</sup>In cylindrical and spherical coordinates, the mass fluxes are

$$\dot{m}_\gamma'' = \frac{1}{R_{\text{out}}} \int_{R_{\text{in}}}^{R_{\text{out}}} \dot{m}_\gamma'''(x) r dr ; \quad \dot{m}_\gamma'' = \frac{1}{R_{\text{out}}^2} \int_{R_{\text{in}}}^{R_{\text{out}}} \dot{m}_\gamma'''(x) r^2 dr \quad (7.39)$$

## 7.2.4 Liquid Fuels

Consider a liquid consisting of one or more components. The evaporation rate of each evaporating liquid component,  $\alpha$ , is governed by the following relation:

$$\dot{m}_\alpha'' = h_m \rho_{\text{film}} \ln(1+B) \left( Y_{\alpha,\text{sv}} + \frac{Y_{\alpha,\text{sv}} - Y_{\alpha,\text{g}}}{B} \right) ; \quad B = \frac{\sum_{\alpha'} (Y_{\alpha',\text{sv}} - Y_{\alpha',\text{g}})}{1 - \sum_{\alpha'} Y_{\alpha',\text{sv}}} \quad (7.42)$$

$h_m$  is the mass transfer coefficient,  $\rho_{\text{film}}$  is the density within a thin surface layer,  $B$  is the Spalding mass transfer number,  $Y_{\alpha,\text{sv}}$  is the “surface vapor” mass fraction of component  $\alpha$  at the liquid surface, and  $Y_{\alpha,\text{g}}$  is the mass fraction of component  $\alpha$  in the center of the grid cell adjacent to the liquid surface. The composition of the surface layer is obtained using Raoult’s law with the Clausius-Clapeyron equation for the equilibrium vapor pressure. The volume fraction of component  $\alpha$  at the surface is:

$$X_{\alpha,\text{sv}} = X_{\alpha,\ell} \exp \left[ -\frac{h_{v,\alpha} W_\alpha}{R} \left( \frac{1}{T_s} - \frac{1}{T_{b,\alpha}} \right) \right] ; \quad Y_{\alpha,\text{sv}} = \frac{X_{\alpha,\text{sv}} W_\alpha}{\sum_{\alpha'} X_{\alpha',\text{sv}} W_{\alpha'} + (1 - \sum_{\alpha'} X_{\alpha',\text{sv}}) W_{\text{air}}} \quad (7.43)$$

Here  $X_{\alpha,\ell}$  is the volume fraction of component  $\alpha$  in the liquid,  $h_{v,\alpha}$  is its heat of vaporization,  $W_\alpha$  is its molecular weight,  $T_{b,\alpha}$  is its boiling temperature, and  $T_s$  is the surface temperature. The mass transfer coefficient is given by

$$h_m = \frac{Sh D_{\text{film}}}{L} ; \quad Sh = 0.037 \text{ Sc}^{\frac{1}{3}} \text{ Re}^{\frac{4}{5}} ; \quad \text{Sc} = \frac{\mu_{\text{film}}}{\rho_{\text{film}} D_{\text{film}}} ; \quad \text{Re} = \frac{\rho_{\text{film}} ||\mathbf{u}|| L}{\mu_{\text{film}}} \quad (7.44)$$

$Sh$  is the Sherwood number and  $D$  is the mole-weighted gas phase diffusivity:

$$D_{\text{film}} = \frac{\sum_{\alpha} X_{\alpha,\text{sv}} D_{\alpha}}{\sum_{\alpha} X_{\alpha,\text{sv}}} \quad (7.45)$$

where  $D_\alpha$  is the diffusivity of species  $\alpha$  into air evaluated at the film temperature. The Reynolds number is based on the gas velocity in the gas cell adjacent to the surface,  $||\mathbf{u}||$ , the length scale,  $L$ , and the density and viscosity of the surface film. The length scale,  $L$ , is 1 m for a liquid pool unless specified otherwise. The film density is

$$\rho_{\text{film}} = \frac{P_0}{R T_{\text{film}} (\sum_{\alpha} Y_{\alpha,\text{sv}} / W_\alpha + (1 - \sum_{\alpha} Y_{\alpha,\text{sv}}) / W_{\text{air}})} \quad (7.46)$$

It is assumed that the composition of the film layer consists of evaporated liquids and air.

In practice, a liquid fuel may have multiple components, like gasoline or kerosene, but for the sake of reducing CPU time, all liquid components can be assumed to evaporate to form a common lumped gas species whose mass fraction is denoted  $Z_g$ . In this case, the effective mass fraction of gas species  $\alpha$  is assumed to be proportional to its value at the liquid surface:

$$Y_{\alpha,\text{g}} = Z_g \frac{Y_{\alpha,\text{sv}}}{\sum_{\alpha'} Y_{\alpha',\text{sv}}} \quad (7.47)$$

The assumption of an assumed lumped gas species is not mandatory—one can evaporate each liquid component to form a unique gas species.

For simplicity, the liquid fuel itself is treated like a thermally-thick solid for the purpose of computing the heat conduction. There is no computation of the convection of the liquid within the pool. Obviously, this assumption has consequences. One of which is related to the fact that the evaporation rate expression (7.42) is applied only to the surface liquid layer. However, it is possible that an interior liquid layer may possess a temperature,  $T_\ell$ , in excess of the boiling temperature of an individual liquid component,  $T_{b,\alpha}$ . If this is the case, an additional amount of component  $\alpha$  is boiled off from the surface so as to drive the liquid

temperature back towards the boiling point of component  $\alpha$ . This addition to the surface mass flux takes the form:

$$\dot{m}_{\alpha}'' = \frac{\rho_{\ell,\alpha} (h(T_{\ell}) - h(T_{b,\alpha})) \delta}{h_{v,\alpha} \delta t} ; \quad h(T) = \int_0^T c_s(T') dT' \quad (7.48)$$

where  $\rho_{\ell,\alpha}$  is the mass per unit volume of liquid component  $\alpha$ ,  $h(T)$  is the enthalpy of the liquid,  $\delta$  is the layer thickness,  $h_{v,\alpha}$  is the heat of vaporization of component  $\alpha$ , and  $\delta t$  is the time step.

## 7.2.5 Shrinking and Swelling Materials

The layer thickness is updated according to the ratio of the instantaneous material density and the density of the material in its pure form. In case of several material components, the amount of swelling and shrinking is determined by the maximum and sum of these ratios, respectively. In each time step, the size of each condensed phase cell is multiplied by the following factor:

$$\delta = \begin{cases} \max_{\alpha} \left( \frac{\rho_{s,\alpha}}{\rho_{\alpha}} \right) & \text{if } \max_{\alpha} \left( \frac{\rho_{s,\alpha}}{\rho_{\alpha}} \right) \geq 1 \\ \sum_{\alpha} \left( \frac{\rho_{s,\alpha}}{\rho_{\alpha}} \right) & \text{if } \max_{\alpha} \left( \frac{\rho_{s,\alpha}}{\rho_{\alpha}} \right) < 1 \end{cases} \quad (7.49)$$

Correspondingly, the densities are divided by the factor  $\delta$  to conserve mass.

# Chapter 8

## Aerosols

A gas species in FDS can be defined as an aerosol species. This enables a series of aerosol behaviors that includes deposition, settling, spray scrubbing, and condensation (for liquid aerosols). This chapter describes the aerosol behaviors in FDS.

### 8.1 Aerosol Deposition

By default, FDS assumes that soot is transported just like all other gaseous species. That is, the soot particles are small enough that their settling velocity is small compared to the fire-driven flows of the gas containing the soot. Near surfaces, however, other mechanisms can affect the soot, which results in its deposition onto surfaces. The removal of soot via deposition can impact the visibility for egress and the time for smoke detectors to activate. In forensic fire reconstructions, the amount of soot deposited on surfaces can be correlated to post-fire observations. The deposition of particulates is also important for computing the dispersion characteristics of aerosol toxicants like ash, radionuclides, or other particulate matter.

However, there is an option to treat any gas phase species as an aerosol that can be deposited on surfaces. Aerosol deposition is determined by applying an empirical deposition velocity to aerosols near surfaces. There are a number of phenomena that cause deposition: thermophoresis (where temperature gradients push the aerosol towards or away from the surface), gravitational settling, diffusive deposition (where the aerosols move along the boundary layer concentration gradient), and turbulent deposition (essentially impact deposition due to a turbulent boundary layer). Other phenomena, such as electrical fields, can also result in deposition but are not considered in FDS due to their relatively small contribution in compartment fire scenarios.

The total aerosol deposition velocity to surfaces,  $u_{\text{dep}}$ , is determined by assuming the deposition phenomena are independent, computing a deposition velocity for each mechanism, and then summing them [79]

$$u_{\text{dep}} = u_g + u_{\text{th}} + u_{\text{dt}} \quad (8.1)$$

where  $u_g$  is the gravitational settling velocity (for cells near upward-facing surfaces),  $u_{\text{th}}$  is the thermophoretic velocity, and  $u_{\text{dt}}$  is the combined diffusion-turbulence velocity. If the aerosol is located in a gas-phase cell adjacent to a wall, then the aerosol (represented by the subscript  $\alpha$ ) is removed from the gas-phase and deposited onto the surface by imposing the following boundary condition

$$\dot{m}_{\text{dep},\alpha}'' = \rho Y_\alpha u_{\text{dep}} \quad (8.2)$$

Using this boundary condition, the aerosol surface density that accumulates on surfaces is tracked, and the amount of aerosol that deposits to a surface is removed from the adjacent gas-phase cell. Note that the subscript  $\alpha$  refers to a species that contains soot or aerosol, whereas the subscript ‘a’ in the remainder of the section refers to the condensed phase soot or aerosol properties, such as mass or density.

### 8.1.1 Gravitational Settling

The gravitational settling velocity is given by [80]

$$u_g = g m_a \frac{C_n}{6\pi \chi_d \mu r_a} \quad (8.3)$$

where  $m_a$  is the particle mass,  $\chi_d$  is a shape factor,  $\mu$  is the dynamic viscosity of air,  $r_a$  is the particle radius, and  $C_n$  is the Cunningham slip correction factor given by [80]

$$C_n = 1 + 1.257 Kn + 0.4 Kn e^{-1.1/Kn} \quad (8.4)$$

where  $Kn$  is the particle Knudsen number given by the ratio of the mean free path of the gas to the particle radius.  $Kn$  is computed as [81]

$$Kn = \frac{\lambda}{r_a} \quad (8.5)$$

where  $\lambda$  is the mean free path of gas molecules which is computed as [82]

$$\lambda = \mu_g \sqrt{\frac{\pi}{2 p \rho_g}} \quad (8.6)$$

For each aerosol species in the gas phase, a gravitational settling velocity is calculated and imposed on the convective term in the species transport equation (Eq. 2.5). This approach is similar to the drift flux model for smoke transport described in Hu et al. [83]. The gravitational settling velocity is also included in the total deposition velocity to deposit aerosols onto upward-facing flat surfaces, as described above.

### 8.1.2 Thermophoretic Deposition

The thermophoretic velocity is computed as

$$u_{th} = \frac{K_{th} v}{T_g} \frac{dT}{dx} \quad (8.7)$$

This requires the wall temperature gradient, which is only resolved in a DNS simulation. For an LES simulation, the temperature gradient is computed from the wall heat transfer coefficient.

$$\frac{dT}{dx} = \frac{h(T_g - T_w)}{k_g} \quad (8.8)$$

$K_{th}$  is the thermophoretic velocity coefficient and is calculated using the following correlation [84]

$$K_{th} = \frac{2 C_s (\alpha + C_t Kn) C_n}{(1 + 3 C_m Kn) (1 + 2 \alpha + 2 C_t Kn)} \quad (8.9)$$

where  $C_s = 1.17$  is the thermal slip coefficient,  $\alpha$  is the ratio of the gas conductivity to the particle conductivity,  $C_m = 1.146$  is the momentum accommodation coefficient, and  $C_t = 2.2$  is the thermal accommodation coefficient.

For each aerosol species in the gas phase, a thermophoretic velocity is calculated and imposed on the convective term in the species transport equation (Eq. 2.5). This approach follows the drift flux model for smoke transport described in Hu et al. [83].

### 8.1.3 Turbulent Deposition

The diffusion-turbulence deposition velocity depends upon the flow regime (diffusion, diffusion-impaction, or inertia-moderated). The deposition velocity for these regimes is given below [85].

$$u_{dt} = \begin{cases} 0.086 \text{ Sc}^{-0.7} u_\tau & \tau^+ < 0.2 \\ 3.5 \times 10^{-4} \tau^{+2} u_\tau & 0.2 < \tau^+ < 22.9 \\ 0.17 u_\tau & \tau^+ > 22.9 \end{cases} \quad (8.10)$$

where  $\text{Sc}$  is the particle Schmidt number, or the ratio of the kinematic viscosity to the Brownian diffusion coefficient of the particle ( $\nu/D_B$ ),  $u_\tau$  is the wall friction velocity computed by the wall model, and  $\tau^+$  is the dimensionless stopping distance given by [86]

$$\tau^+ = \frac{\rho_a (2r_a)^2}{18 \mu^2} u_\tau^2 \rho_g \quad (8.11)$$

## 8.2 Aerosol Agglomeration

Agglomeration is the process via which small aerosol particles become larger aerosol particles due to collisions that cause the smaller particles to stick together and become larger particles. The FDS implementation of aerosol agglomeration is based on the agglomeration model in VICTORIA [87], a US NRC software package for modeling radioactive aerosol transport during severe accidents. The agglomeration mechanisms in FDS include Brownian, gravitational, and turbulent (shear and inertial). The agglomeration is governed by the equation below.

$$\frac{dN}{dt} = \frac{1}{2} \int_0^m \Phi(\omega, m - \omega) N(\omega) N(m - \omega) d\omega - N(m) \int_0^\infty \Phi(\omega, m) N(\omega) d\omega - R(m) + S(m) \quad (8.12)$$

In the above equation  $\Phi$  is an agglomeration kernel,  $N(m)$  is the number density of particles of size  $m$ ,  $R$  is a removal term (e.g. outflow, oxidation), and  $S$  is a source term (e.g. from combustion). The equation states that the rate of formation of particles of size  $m$  is the rate that two particles whose combined size is  $m$  collide, minus the rate at which particles of size  $m$  collide with other particles and become a larger size, minus removal of particles of size  $m$ , and plus the generation of particles of size  $m$ . The selected approach for solving the agglomeration equation is to bin the particle sizes into multiple bins and transform the integrals over particle size into summations over particle bins. By defining the maximum,  $m_{max}$ , and minimum,  $m_{min}$ , particle mass and the number of bins,  $M$ , one can define the average bin mass,  $x$ , as shown below [88].

$$s = \left( \frac{m_{max}}{m_{min}} \right)^{1/M}, \quad m_i = s m_{i-1}, \quad x_i = \frac{2m_i}{1+s} \quad (8.13)$$

Using this binning, the agglomeration equation becomes:

$$\frac{dN_i}{dt} = \sum_{k=1}^M \sum_{j=k}^M \left( 1 - \frac{1}{2} \delta_{j,k} \right) \eta \Phi(j, k) N_j N_k - N_i \sum_{k=1}^M \Phi(j, k) N_k - R_i + S_i \quad (8.14)$$

where  $\eta$ , shown below, is a function for apportioning mass between adjacent particle bins.

$$\eta = \begin{cases} \frac{x_{i+1} - m_i}{x_{i+1} - x_i} & x_i < m_i \leq x_{i+1} \\ \frac{m_i - x_{i-1}}{x_i - x_{i-1}} & x_{i-1} < m_i \leq x_i \end{cases} \quad (8.15)$$

The agglomeration kernel is given by the sum of the Brownian (Br), gravitational (Gr), shear (Sh), and inertial (In) agglomeration terms.

$$\Phi(m, \omega) = \Phi_{Br}(m, \omega) + \Phi_{Gr}(m, \omega) + \sqrt{\Phi_{Sh}(m, \omega)^2 + \Phi_{In}(m, \omega)^2} \quad (8.16)$$

Brownian agglomeration occurs when the random walks of particles brings them into contact with one another. It is computed as [87]:

$$\Phi_{Br} = 4\pi k_b T (B(m) + B(\omega)) (r_m + r_\omega) Fu(m, \omega) \quad (8.17)$$

where  $Fu$  is the Fuchs factor and  $B$  is the particle mobility.

$$B(m) = \frac{Cn}{6\pi\mu r_m} \quad (8.18)$$

$$\frac{1}{Fu(m, \omega)} = \left( \epsilon_s \frac{r_m + r_\omega}{k_b T (B(m) + B(\omega))} \sqrt{\frac{8k_b T}{\pi} \left( \frac{1}{m} + \frac{1}{\omega} \right)} \right)^{-1} + \left( 1 + \frac{2\sqrt{(\tilde{a}_m^2 + \tilde{a}_\omega^2)}}{r_m + r_\omega} \right)^{-1} \quad (8.19)$$

$$\tilde{a}_m = \frac{(r_m + a_m)^3 - (r_m^2 + a_m^2)^{3/2}}{3r_m a_m} - r_m \quad (8.20)$$

$$a_m = B(m) \sqrt{\frac{2k_b T m}{\pi}} \quad (8.21)$$

Gravitational agglomeration results from heavier particle falling onto larger particles. The efficiency of these collisions are governed by a sticking factor,  $\varepsilon_S$  assumed to be one, and a collision efficiency,  $\varepsilon_{PK}$  [87]

$$\Phi_{Gr}(m, \omega) = \varepsilon_S \varepsilon_{PK}(m, \omega) (r_m + r_\omega)^2 |u_g(m) - u_g(\omega)| \quad (8.22)$$

$$\varepsilon_{PK}(m, \omega) = \frac{\min(r_m, r_\omega)^2}{2(r_m + r_\omega)^2} \quad (8.23)$$

$$u_g(m) = g m B(m) \quad (8.24)$$

Shear agglomeration results from particles of different speeds moving along streamlines that bring them into contact with one another [87].

$$\Phi_{Sh}(m, \omega) = \varepsilon_S \varepsilon_{PK}(m, \omega) \chi_C^3 (r_m + r_\omega)^3 \sqrt{\frac{8\rho\pi\varepsilon}{15\mu}} \quad (8.25)$$

where  $\chi_C$  is a shape factor, assumed to be one, and  $\varepsilon$  is the turbulent kinetic energy dissipation rate.

Inertial agglomeration results from particles departing from the flow streamlines due to their inertia and coming into contact with other particles [87].

$$\Phi_{In}(m, \omega) = \varepsilon_S \varepsilon_{PK}(m, \omega) \chi_C^2 (r_m + r_\omega)^2 \left( \frac{512\rho\pi^3\varepsilon^3}{15\mu} \right)^{1/4} \frac{|u_g(m) - u_g(\omega)|}{g} \quad (8.26)$$

### 8.3 Aerosol Scrubbing

Scrubbing is the removal of aerosols from the gas phase. In FDS, aerosols can be scrubbed by water sprays. The correlations used to define the scrubbing efficiency are based on the containment spray scrubbing model in MELCOR [89]. The rate of aerosol scrubbing in a grid cell is based on the volume of the grid cell swept by a droplet over a timestep and the removal efficiency,  $\varepsilon$ , for that droplet. The fraction of a grid cell's volume swept during a time step,  $F$ , is determined by the projected surface area of the droplet and its velocity:

$$F = \frac{\min(C\pi r_p^2(\delta x_i \delta y_j \delta z_k)^{2/3}) u_p \Delta t_p}{\delta x_i \delta y_j \delta z_k} \quad (8.27)$$

where  $r_p$  is the droplet radius,  $C$  is the weighting factor for the droplet,  $u_p$  is the droplet velocity, and  $\Delta t_p$  is the time step for the current path segment of the droplet (droplet movement may be evaluated using a smaller time steps than the global time step,  $\Delta t$  if the droplet will cross a cell boundary over the global time step).

The removal efficiency is given by the product sum of an interception efficiency,  $\varepsilon_{in}$ , and an impaction efficiency,  $\varepsilon_{im}$ . Both result from summing viscous, *vis*, and potential flow, *pot*, behaviors.

$$\varepsilon_{in,vis} = \left(1 + \frac{r_a}{r_p}\right)^2 \left(1 - \frac{3}{2\left(1 + \frac{r_a}{r_p}\right)} + \frac{1}{2\left(1 + \frac{r_a}{r_p}\right)^3}\right) \quad (8.28)$$

$$\varepsilon_{in,pot} = \left(1 + \frac{r_a}{r_p}\right)^2 - \left(1 + \frac{r_a}{r_p}\right) \quad (8.29)$$

$$\varepsilon_{im,vis} = \begin{cases} 0 & Stk \leq 1.214 \\ \left(1 + \frac{0.75 \ln(2Stk)}{Stk - 1.214}\right)^{-2} & Stk > 1.214 \end{cases} \quad (8.30)$$

$$\varepsilon_{im,pot} = \begin{cases} 0 & Stk \leq 0.0834 \\ \left(\frac{Stk}{Stk + 0.5}\right)^2 & Stk > 0.2 \end{cases} \quad (8.31)$$

Where  $Stk$  is the Stokes number. For other values of  $Stk$ ,  $\varepsilon_{im,pot}$  is interpolated. The viscous and potential values are combined as:

$$\varepsilon_x = \frac{\varepsilon_{x,vis} + \varepsilon_{x,pot}(\text{Re}_p/60)}{1 + (\text{Re}_p/60)} \quad (8.32)$$

The total efficiency is given as:

$$\varepsilon = 1 - (1 - \varepsilon_{in})(1 - \varepsilon_{im}) \quad (8.33)$$

The rate of removal is then given by:

$$\lambda = \frac{F \varepsilon Y_a \rho}{\Delta t} \quad (8.34)$$

## 8.4 Aerosol Condensation

If a gas species with liquid properties is defined as an aerosol, FDS will compute condensation and evaporation for that species. This is accomplished by defining two tracked species: one non-aerosol species for the gas phase, and a second aerosol species for the condensed phase. Condensation and evaporation will be computed both in the gas and on any solid surface in the domain.

### 8.4.1 Gas Condensation

If a condensable vapor species is present in a gas cell, a check is made on the equilibrium vapor fraction of the cell using Eq. (9.30) where  $T_p$  is assumed to be the cell temperature. If the mass fraction in the cell exceeds the equilibrium fraction then condensation will occur. If the mass fraction in the cell is less than the equilibrium mass fraction, then evaporation will occur.

The evaporation rate is computed using the mass transfer coefficient as defined in Eq (9.31) through Eq (9.32). The surface area for evaporation is determined by the total number of condensed water droplets in the cell which is computed from the mass of water vapor present and the user defined particle diameter for the aerosol species.

The condensation rate is computed assuming the same mass transfer coefficient as evaporation. Since it is possible that there is no existing droplets to condense on, a cell with no existing condensed vapor uses an assumed concentration of nucleation sites/cm<sup>3</sup> that are assumed to have the same diameter as the condensed species.

The total mass evaporated or condensed is added to the bulk mass source terms,  $\dot{m}_{b,\alpha}'''$ , for the condensed and vapor phase species. The bulk energy source term,  $\dot{q}_b'''$ , is taken as the mass source term multiplied by the heat of vaporization,  $h_v$  where condensation results in positive energy source term and evaporation results in a negative energy source term.

The contribution to divergence is added to the D\_SOURCE term following Eq (9.36). Since the condensed and vapor phases have the same molecular weight, the first term in the equation is zero.

For the condensed phase, the radiation absorption is computed following Sec. 6.3.

### 8.4.2 Wall Condensation

Condensation and evaporation for a wall cell follows a similar process to a gas cell. The process begins with an equilibrium vapor check where  $T_b$  is assumed to be the wall temperature. The rate of evaporation and condensation is based on a wall function [90].

$$Y_\alpha^+ = \frac{\rho u_\tau}{\dot{m}_{dep,\alpha}'''} (Y_\alpha - Y_{\alpha,\ell}) \quad (8.35)$$

$$y^+ = \frac{\rho u_\tau \delta n / 2}{\mu} \quad (8.36)$$

$$Y_\alpha^+ = Sc y^+ e^{-\Gamma} + (2.12 \ln(y^+) + (3.85 Sc^{1/3} - 1.3)^2 + 2.12 \ln(Sc)) e^{-1/\Gamma} \quad (8.37)$$

$$\Gamma = \frac{0.01 (Sc y^+)^4}{1 + 5 Sc^3 y^+} \quad (8.38)$$

The total mass evaporated or condensed is added to the deposition wall mass source terms,  $\dot{m}_{dep,\alpha}''$ . The energy required is added to a wall condensation energy source term that accounts for the heat of vaporization and the enthalpy change going from the gas temperature to the wall temperature.

$$\dot{q}_{dep,\alpha}'' = \dot{m}_{dep,\alpha}'' (h_v(T_w) + h_{s,\alpha}(T_g) - h_{s,\alpha}(T_w)) \quad (8.39)$$

For the D\_SOURCE term only the first term in Eq (9.36) applies since all the phase change energy comes from the wall cell.

# Chapter 9

## Lagrangian Particles

Lagrangian particles are used to represent a wide variety of objects that cannot be resolved on the numerical grid. Liquid droplets are the most common example. This chapter outlines the treatment of the transport; size distribution; and mass, momentum and energy transfer to and from Lagrangian particles. The formulation presented here closely follows the dispersed discrete-element formulation presented in [91].

### 9.1 Particle Transport in the Gas Phase

In the gas phase momentum equation, Eq. (4.36), the force term  $\mathbf{f}_b$  represents the momentum transferred from particles to the gas. It is obtained by summing the force transferred from each particle in a grid cell and dividing by the cell volume,  $V$ :

$$\mathbf{f}_b = \frac{1}{V} \sum \left[ \frac{\rho}{2} C_d A_{p,c} (\mathbf{u}_p - \mathbf{u}) |\mathbf{u}_p - \mathbf{u}| - \frac{dm_p}{dt} (\mathbf{u}_p - \mathbf{u}) \right] \quad (9.1)$$

where  $C_d$  is the drag coefficient,  $A_{p,c}$  is the particle cross-sectional area,  $r_p$  is the particle radius,  $\mathbf{u}_p$  is the particle velocity,  $m_p$  is the particle mass,  $\mathbf{u}$  is the gas velocity, and  $\rho$  is the gas density. The subscript “b” stands for “bulk,” meaning that the particles in the cell represent a bulk mass dragging on the gas.

The particle acceleration is given by

$$\frac{d\mathbf{u}_p}{dt} = \mathbf{g} - \frac{1}{2} \frac{\rho C_d A_{p,c}}{m_p} (\mathbf{u}_p - \mathbf{u}) |\mathbf{u}_p - \mathbf{u}| \quad (9.2)$$

The particle position,  $\mathbf{x}_p$ , is determined from the equation

$$\frac{d\mathbf{x}_p}{dt} = \mathbf{u}_p \quad (9.3)$$

The exact solution procedure of the above model is presented in Appendix I. The drag coefficient (default based on a sphere) is a function of the local Reynolds number that is based on the particle diameter,  $D$  ( $2r_p$ )

$$C_d = \begin{cases} 24/\text{Re}_D & \text{Re}_D < 1 \\ 24 (0.85 + 0.15 \text{Re}_D^{0.687}) / \text{Re}_D & 1 < \text{Re}_D < 1000 \\ 0.44 & 1000 < \text{Re}_D \end{cases} \quad (9.4)$$

$$\text{Re}_D = \frac{\rho |\mathbf{u}_p - \mathbf{u}| 2r_p}{\mu(T)} \quad (9.5)$$

where  $\mu(T)$  is the dynamic viscosity of air at temperature  $T$  [91]<sup>1</sup>. For cylindrical particles, the following correlation was derived from data presented by Schlichting [92]:

$$C_d = \begin{cases} 10/\text{Re}_D^{0.8} & \text{Re}_D < 1 \\ 10(0.6 + 0.4\text{Re}_D^{0.8})/\text{Re}_D & 1 < \text{Re}_D < 1000 \\ 1 & 1000 < \text{Re}_D \end{cases} \quad (9.6)$$

For thin disk shaped particles, the following correlation was derived from data presented in Hoerner [93]:

$$C_d = \frac{20.37}{\text{Re}_D} + \frac{1.17}{1 + 1/\text{Re}_D}, \quad (9.7)$$

where the particle diameter,  $D$ , for square cartesian particles is taken as the hydraulic diameter,  $A_{p,c}^{1/2}$ .

Additional corrections are made to account for drag reduction due to the wake effect [94] and deformation of the droplet [95].

### 9.1.1 Using Lagrangian Particles to Represent Vegetation

For some applications, it is convenient to use Lagrangian particles to represent stationary vegetation, like grasses or leaves, or airborne vegetation, like burning brands. Typically, these different types of vegetation are grouped into different classes of fuel “elements”, each of which is represented by a single representative particle in each grid cell. Each representative particle contributes to the bulk force term:

$$\mathbf{f}_b = \sum_e \mathbf{f}_{b,e} \quad ; \quad \mathbf{f}_{b,e} = \frac{\rho}{2} C_d C_{s,e} \beta_e \sigma_e (\mathbf{u}_{p,e} - \mathbf{u}) |\mathbf{u}_{p,e} - \mathbf{u}| + \dot{m}_{p,e}''' (\mathbf{u}_{p,e} - \mathbf{u}) \quad (9.8)$$

The terms with subscript “e” refer to a particular class of fuel elements and are determined geometrically or empirically. The shape factor,  $C_{s,e}$ , is the ratio of the particle’s projected cross sectional area,  $A_{p,c}$ , to its surface area,  $A_{p,s}$ . A perfect sphere has a shape factor of  $\pi r^2 / 4\pi r^2 = 1/4$ , but for actual vegetation, the shape factor is assigned an empirical value that accounts for both shape and orientation. Pine needles, for example, project a different cross sectional area depending on their orientation. The drag coefficient,  $C_d$ , is empirically derived and takes into account both the geometry and shadowing effects of closely packed objects. It is typically expressed as a constant rather than a function of the Reynolds number.  $\beta_e$  is the volume fraction; that is, the ratio of the volume occupied by solid mass to the overall volume of the vegetation, sometimes referred to as the *packing ratio*.  $\sigma_e$  is the surface to volume ratio of a single particle.  $\dot{m}_{p,e}'''$  is the mass generation rate per unit volume of that particular vegetation element.

A convenient way to describe the geometric properties of vegetation is by specifying the surface to volume ratio,  $\sigma_e$ , the volume (packing) ratio,  $\beta_e$ , and an assumed shape, like a sphere or cylinder. The volume fraction is sometimes expressed as a mass per unit volume, or *bulk density*,  $m''' = \rho_e \beta_e$ , where  $\rho_e$  is the density of the solid. With this information, and the following relations:

$$C_{s,e} = \frac{A_{p,c}}{A_{p,s}} \quad ; \quad \beta_e = \frac{n_{p,e} V_p}{V} = \frac{m'''}{\rho_e} \quad ; \quad \sigma_e = \frac{A_{p,s}}{V_p} \quad (9.9)$$

we can convert the drag force expression in Eq. (9.8) to its equivalent in Eq. (9.1) where the single particle element is represented as  $n_{p,e}$  spheres or cylinders occupying a grid cell volume,  $V$ . In the model, it is sufficient to have only one weighted particle per grid cell per fuel element to represent all of the actual particles of that vegetation class.

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<sup>1</sup>Note that the 0.85 in the second line of Eq. (9.4) has been inserted to ensure continuity of the functional relationship at  $\text{Re}_D = 1$ .

One further simplification of Eq. (9.8) is made by lumping terms into a single *frontal area density*

$$\kappa_e = C_{s,e} \beta_e \sigma_e = \frac{n_{p,e} A_{p,c}}{V} \quad (9.10)$$

$\kappa_e$  can also be thought of as an *absorption coefficient* for the purpose of computing thermal radiation absorption. In practice,  $\kappa_e$  is not determined from Eq. (9.10). Rather, it is obtained by measuring the relative amount of sunlight that penetrates a slab of vegetation of thickness,  $L$ :

$$W = e^{-\kappa_e L} \quad (9.11)$$

Here,  $W$  is the relative area of “white” of the shadowgraph formed by sunlight passing through the vegetation. It can also be thought of as the fraction of blue sky one sees when looking through a tree canopy of thickness  $L$ . It is often referred to as the *frontal area index* or *frontal area fraction*, not to be confused with the *frontal area density*,  $\kappa_e$ .

### 9.1.2 Drag Reduction

Typically, Lagrangian particle models assume that the particles are independent of each other. If the spray is dense enough, however, the individual particles influence each other through aerodynamic interactions. These effects cannot be captured by the current Eulerian-Lagrangian model for two reasons. First, the Lagrangian particles occupy no volume in the Eulerian space. Second, the separation lengths would be of sub-grid scale in most practical simulations. The aerodynamic interactions start to have an effect when the average particle spacing is less than 10 diameters [96, 97]. This corresponds to a particle volume fraction,  $\alpha$ , of approximately 0.01. Volume fractions as high as this can sometimes be achieved inside water mist sprays. If the spray is even more dense, particle-particle collisions or four-way coupling would need to be considered.

In a configuration where two particles with the same diameter are directly in line, the reduction of the drag force on the second particle is assumed to be: [94]:

$$C_d = C_{d,0} \frac{F}{F_0} \quad (9.12)$$

where  $C_{d,0}$  is the single particle drag coefficient and  $F/F_0$  is the hydrodynamic force ratio of the trailing particle to an isolated particle:

$$\frac{F}{F_0} = W \left[ 1 + \frac{\text{Re}}{16} \frac{1}{(L/D - 1/2)^2} \exp \left( -\frac{\text{Re}}{16} \frac{1}{(L/D - 1/2)} \right) \right] \quad (9.13)$$

where  $\text{Re}$  is the single particle Reynolds number,  $L$  is the distance between the particle centers, and  $W$  is the non-dimensional, non-disturbed wake velocity at the center of the trailing particle

$$W = 1 - \frac{C_{d,0}}{2} \left[ 1 - \exp \left( -\frac{\text{Re}}{16} \frac{1}{(L/D - 1/2)} \right) \right] \quad (9.14)$$

The average separation distance between particle centers,  $L/D$ , is calculated from the local particle volume fraction,  $\alpha$ , as follows:

$$L/D = (\pi/6\alpha)^{\frac{1}{3}} \quad (9.15)$$

This simple approximation assumes that particles are uniformly distributed in each computational cell [98]. In reality, the spray is not monodisperse and the separation distance between the interacting particles varies.

In the simulation, the drag reduction factor in Eq. 9.12 is only used when the local droplet volume fraction exceeds  $1 \times 10^{-5}$ .

An alternative approach to drag reduction was provided by Prahl et al. [96] who studied the interaction between two solid spheres in steady or pulsating flow by detailed numerical simulations. According to their study, the above equation underestimates the drag reduction significantly at small drop-to-drop distances. The inflow pulsations were found to reduce the effect of the drag reduction. At large distances, the two results are similar, the Ramírez-Muñoz correlation showing more drag reduction. This is not surprising since the velocity profile of a fully developed axi-symmetric wake behind an axi-symmetric body is used in developing the drag reduction correction in Eqs. 9.13 and 9.14. At short distances, the wake is not fully developed and the assumption does not hold.

## 9.2 Liquid Droplet Size Distribution

The cumulative volume distribution for a liquid spray is represented by a combination of log-normal and Rosin-Rammler distributions [99]:

$$F_v(D) = \begin{cases} \frac{1}{\sqrt{2\pi}} \int_0^D \frac{1}{\sigma D'} \exp\left(-\frac{[\ln(D'/D_{v,0.5})]^2}{2\sigma^2}\right) dD' & (D \leq D_{v,0.5}) \\ 1 - \exp\left(-0.693\left(\frac{D}{D_{v,0.5}}\right)^\gamma\right) & (D > D_{v,0.5}) \end{cases} \quad (9.16)$$

where  $D_{v,0.5}$  is the median volumetric droplet diameter (i.e., half the mass is carried by droplets with diameters of  $D_{v,0.5}$  or less), and  $\gamma$  and  $\sigma$  are empirical constants equal to approximately 2.4 and 0.48, respectively.<sup>2</sup> Alternatively, the user may specify any form of size distribution using tabulated input data.

The median droplet diameter is a function of the sprinkler orifice diameter, operating pressure, and geometry. Research at Factory Mutual has yielded a correlation for the median droplet diameter [100]

$$\frac{D_{v,0.5}}{d} \propto We^{-\frac{1}{3}} \quad (9.17)$$

where  $d$  is the orifice diameter of the nozzle. The orifice Weber number, the ratio of inertial forces to surface tension forces, is given by

$$We = \frac{\rho_p u_p^2 d}{\sigma} \quad (9.18)$$

where  $\rho_p$  is the liquid density,  $u_p$  is the discharge velocity, and  $\sigma$  is the liquid surface tension ( $72.8 \times 10^{-3}$  N/m at  $20^\circ\text{C}$  for water). The discharge velocity can be computed from the mass flow rate, a function of the operating pressure and orifice coefficient known as the K-factor. FM reports that the constant of proportionality in Eq. (9.17) appears to be independent of flow rate and operating pressure. Three different sprinklers were tested in their study with orifice diameters of 16.3 mm, 13.5 mm, and 12.7 mm, and the constants were approximately 4.3, 2.9, and 2.3, respectively. The strike plates of the two smaller sprinklers were notched, while that of the largest sprinkler was not [100].

In real sprinkler systems, the operating pressure is affected by the number of open nozzles. Typically, the pressure in the piping is high when the first sprinkler activates, and decreases when more and more sprinkler heads are activated. The pipe pressure has an effect on flow rate, droplet velocity and droplet size distribution. FDS does not predict the variation of pipe pressure; it should be specified by the user. The following dependencies are used to update the droplet boundary conditions for mass flow, droplet speed, and median diameter:

$$\dot{m}_p \propto p^{1/2} \quad ; \quad u_p \propto p^{1/2} \quad ; \quad D_{v,0.5} \propto p^{-1/3} \quad (9.19)$$

---

<sup>2</sup>The Rosin-Rammler and log-normal distributions are smoothly joined if  $\sigma = 2/(\sqrt{2\pi}(\ln 2)^\gamma) = 1.15/\gamma$ .

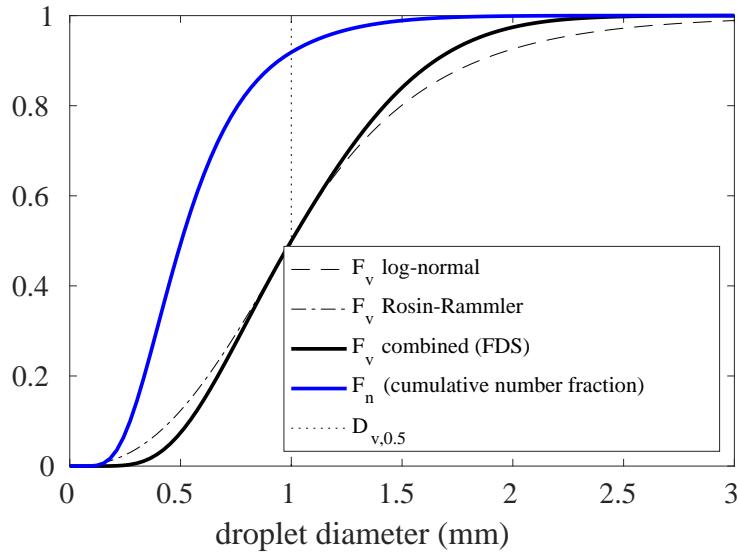


Figure 9.1: Cumulative Volume Fraction and Cumulative Number Fraction functions of the droplet size distribution from a typical industrial-scale sprinkler. The median volumetric diameter,  $D_{v,0.5}$ , is 1 mm,  $\sigma = 0.48$  and  $\gamma = 2.4$ .

The droplet diameters are randomly chosen from the given size distribution. The cumulative number fraction (CNF),  $F_n$ , is determined from the cumulative volume fraction,  $F_v$ , as follows

$$F_n(D) = \int_0^D \frac{F'_v(D')}{D'^3} dD' \Big/ \int_0^\infty \frac{F'_v(D')}{D'^3} dD' ; \quad F'_v \equiv \frac{dF_v}{dD} \quad (9.20)$$

Figure 9.1 displays the Rosin-Rammler/log-normal function and the resulting cumulative number fraction.

The selection of droplet diameters makes use of a stratified sampling technique to ensure that the droplets span the entire range of sizes, even with a relatively small number of droplets. Without the stratification, the tails of the distribution can be poorly represented. The procedure for selecting droplet sizes is as follows:

1. Suppose that the mass flow rate of the liquid is  $\dot{m}$ , that the time interval for droplet insertion is  $\delta t$ , and that the number of droplets inserted each time interval is  $N$ .
2. Divide the droplet diameter range into a number of bins of equal width.
3. Randomly choose  $N$  integers,  $n_i$ , ranging from 1 to the total number of bins.
4. Choose  $N$  uniformly distributed real numbers between 0 and 1 and calculate  $N$  random droplet diameters:

$$D_i = F_n^{-1} \left[ F_n(D_{n_i,\min}) + \mathcal{U}(0, 1) (F_n(D_{n_i,\max}) - F_n(D_{n_i,\min})) \right] \quad (9.21)$$

where  $\mathcal{U}(0, 1)$  is a uniformly distributed real number between 0 and 1 and  $D_{n_i,\min}$  and  $D_{n_i,\max}$  are the minimum and maximum diameters of bin  $n_i$ .

5. Compute weighting constants for each droplet  $C_i = F_n(D_{n_i,\max}) - F_n(D_{n_i,\min})$ .

6. Compute a global weighting constant, C, to maintain the overall mass balance:

$$\dot{m} \delta t = C \sum_{i=1}^N C_i \frac{4}{3} \pi \rho_p \left( \frac{D_i}{2} \right)^3 \quad (9.22)$$

The mass and heat transferred from each droplet will be multiplied by the weighting factor  $C$ .

### 9.3 Spray Initialization

The droplets are introduced into the simulation along a spherical surface whose diameter is a specified stand-off distance from the nozzle orifice. It is assumed that the droplets have fully atomized by this stage. The longitude of the initial droplet position,  $0 \leq \theta < 2\pi$ , is randomly chosen from a uniform distribution. The latitude,  $0 \leq \phi < \pi$ , is randomly selected from the following distribution:

$$f(\phi) = \exp \left[ -\beta \left( \frac{\phi - \mu}{\phi_{\max} - \phi_{\min}} \right)^2 \right] \quad (9.23)$$

Note that  $\phi = 0$  is the south pole of the sphere. The parameter  $\phi_{\min} = 0$  by default. The parameter  $\mu$  controls the location of the peak of the distribution and may be used to approximate a hollow cone nozzle. By default,  $\mu = \phi_{\min} = 0$ . However, if  $\phi_{\min} > 0$ , unless otherwise specified,  $\mu$  is set to the average of  $\phi_{\min}$  and  $\phi_{\max}$ . The spread parameter,  $\beta$ , is 5 by default. If  $\beta = 0$  the distribution is uniform. As  $\beta$  becomes large, the distribution approximates a delta function at  $\mu$ . All the droplets are given the same initial speed in the direction of the surface normal.

### 9.4 Heating and Evaporation of Liquid Droplets

Liquid droplets can represent either discrete airborne spheres or elements of the thin liquid film that forms on wetted solids. These film “droplets” are still individually tracked as Lagrangian particles, but the heat and mass transfer coefficients are different. In the discussion to follow, the term “droplets” will be used to describe either form.

Over the course of a time step of the gas phase solver, the droplets in a given grid cell evaporate to form the gas species  $\alpha$ . The evaporation rate is a function of the liquid equilibrium vapor mass fraction,  $Y_{\alpha,\ell}$ , the local gas phase vapor mass fraction,  $Y_{\alpha,g}$ , the (assumed uniform) droplet temperature,  $T_p$ , and the local gas temperature,  $T_g$ . The subscript “g” refers to the average of the quantity in the cell occupied by the droplet; “p” refers to the liquid droplet; and if the droplet is attached to a solid surface, “s” refers to the surface with temperature  $T_s$ . The mass and energy transfer between the gas, liquid, and possibly a solid surface can be described by the following set of equations [101]

$$\frac{dm_p}{dt} = -A_p h_m \rho_f (Y_{\alpha,\ell} - Y_{\alpha,g}) \quad (9.24)$$

$$\rho_g V \frac{dY_{\alpha,g}}{dt} = - (1 - Y_{\alpha,g}) \frac{dm_p}{dt} \quad (9.25)$$

$$\frac{dT_p}{dt} = \frac{1}{m_p c_p} \left[ \dot{q}_r + A_p h_g (T_g - T_p) + A_p h_s (T_s - T_p) + \frac{dm_p}{dt} h_v \right] \quad (9.26)$$

$$\frac{dT_g}{dt} = \frac{1}{m_g c_g} \left[ A_p h_g (T_p - T_g) - \frac{dm_p}{dt} (h_{\alpha,p} - h_{\alpha,g}) \right] \quad (9.27)$$

$$\frac{dT_s}{dt} = -\frac{A_p h_s}{m_s c_s} (T_s - T_p) \quad (9.28)$$

The droplet is taken to be a pure liquid of species  $\alpha$  (usually, either water or fuel) with a heat of vaporization  $h_v$  and mass  $m_p$ . The mass transfer coefficient,  $h_m$ , is discussed below;  $\rho_g$  is the gas density;  $\rho_f$  is the particle film density;  $c_p$  is the liquid specific heat;  $c_g$  is the gas specific heat;  $h_g$  is the heat transfer coefficient between the droplet and the gas;  $h_s$  is the heat transfer coefficient between the droplet and the solid;  $\dot{q}_r$  is the rate of radiative heating of the droplet (see Eq. (9.41));  $h_{\alpha,p}$  is the vapor specific enthalpy at the particle temperature; and  $h_{\alpha,g}$  is the vapor specific enthalpy at the gas temperature. The surface area of the droplet depends on whether it is airborne or forms part of the solid surface film layer:

$$A_p = \begin{cases} 4\pi r_p^2 & \text{airborne droplet} \\ m_p/(\delta\rho_p) & \text{surface film} \end{cases} \quad (9.29)$$

where  $\delta$  is the thickness of the liquid layer on the solid surface. Equation (9.28) applies only if the droplet is attached to a solid surface in which case the droplet makes up a fraction of the thin film layer with the area given in Eq. (9.29) exposed to the gas and the same area exposed to the solid. The term  $m_s$  in Eq. (9.28) is the mass of the surface layer of the solid, and  $c_s$  is the solid specific heat.

The vapor mass fraction of the gas,  $Y_{\alpha,g}$ , is obtained from the gas phase mass transport equations, and the liquid equilibrium vapor mass fraction is obtained from the Clausius-Clapeyron equation

$$X_{\alpha,\ell} = \exp \left[ \frac{h_v W_\alpha}{R} \left( \frac{1}{T_b} - \frac{1}{T_p} \right) \right] ; \quad Y_{\alpha,\ell} = \frac{X_{\alpha,\ell}}{X_{\alpha,\ell}(1 - W_a/W_\alpha) + W_a/W_\alpha} \quad (9.30)$$

where  $X_{\alpha,\ell}$  is the equilibrium vapor *volume* fraction,  $W_\alpha$  is the molecular weight of the gaseous species  $\alpha$ ,  $W_a$  is the molecular weight of air,  $R$  is the universal gas constant, and  $T_b$  is the boiling temperature of the liquid at standard atmospheric pressure.

Mass and heat transfer between liquid and gas are described with analogous empirical correlations. The mass transfer coefficient,  $h_m$ , is described by the empirical relationships [102, 75]:

$$h_m = \frac{\text{Sh} D_{\ell g}}{L} \frac{B_M}{Y_{\alpha,\ell} - Y_{\alpha,g}} ; \quad B_M = \frac{Y_{\alpha,\ell} - Y_{\alpha,g}}{1 - Y_{\alpha,\ell}} \quad (9.31)$$

$$\text{Sh} = \frac{\ln(1 + B_M)}{B_M} \text{Sh}_0 ; \quad \text{Sh}_0 = \begin{cases} 2 + 0.6 \text{Re}_D^{\frac{1}{2}} \text{Sc}^{\frac{1}{3}} & \text{airborne droplet} \\ 0.0296 \text{Re}_L^{\frac{4}{5}} \text{Sc}^{\frac{1}{3}} & \text{surface film} \end{cases} \quad (9.32)$$

where  $B_M$  is the Spalding mass transfer number [103],  $\text{Sh}_0$  is the low mass flux Sherwood number,  $\text{Sh}$  is the Sherwood number with blowing,  $D_{\ell g}$  is the binary diffusion coefficient between the liquid vapor and the surrounding gas (usually assumed air),  $L$  is a length scale equal to either the droplet diameter or 1 m for a surface film,  $\text{Re}_D$  is the Reynolds number of the droplet (based on the diameter,  $D$ , and the relative air-droplet velocity),  $\text{Re}_L$  is the Reynolds number based on the length scale  $L$ , and  $\text{Sc}$  is the Schmidt number ( $v/D_{\ell g}$ , assumed 0.6 for all cases). Note that the mass transfer coefficient,  $h_m$ , is formulated so that the mass flux model given by Eq. (9.24) recovers Eq. (108) of Sazhin [102] but still works within the implicit time update.

An analogous relationship exists for the heat transfer coefficient between the droplet and the gas [102]:

$$h_g = \frac{\text{Nu} k_g}{L} ; \quad \text{Nu} = \frac{\ln(1 + B_M)}{B_M} \text{Nu}_0 ; \quad \text{Nu}_0 = \begin{cases} 2 + 0.6 \text{Re}_D^{\frac{1}{2}} \text{Pr}^{\frac{1}{3}} & \text{airborne droplet} \\ 0.0296 \text{Re}_L^{\frac{4}{5}} \text{Pr}^{\frac{1}{3}} & \text{surface film} \end{cases} \quad (9.33)$$

where  $\text{Nu}_0$  is the low mass flux Nusselt number,  $\text{Nu}$  is the Nusselt number with blowing,  $k_g$  is the thermal conductivity of the gas, and  $\text{Pr}$  is the molecular Prandtl number,  $\text{Pr} = \mu_f c_{p,f}/k_f$ , with properties evaluated at

the film temperature,  $T_f = T_s + \frac{1}{3}(T_g - T_s)$ . The length scale,  $L$ , is the diameter,  $D$ , of the droplet if airborne, and 1 m if the droplet forms a liquid surface layer.

The heat transfer coefficient between the droplet and a solid surface,  $h_s$ , is either a user-specified constant or an empirical function [75]:

$$h_s = \max \left( 100, 0.0296 \text{Re}_L^{\frac{4}{5}} \text{Pr}^{\frac{1}{3}} \frac{k_p}{L} \right) ; \quad \text{Re}_L = \frac{\rho_p \|\mathbf{u}_p\| L}{\mu_p} \quad (9.34)$$

The velocity of the surface liquid,  $\|\mathbf{u}_p\|$ , is taken as the user-specified constant droplet speed on either a horizontal or vertical surface. The length,  $L$ , is taken as 1 m. The minimum value of 100 W/(m<sup>2</sup>·K) is an order of magnitude estimate for cases where the user has specified a very low droplet velocity on the solid surface.

The exchange of mass and energy between liquid droplets and the surrounding gases (or solid surfaces) is computed droplet by droplet. After the temperature of each droplet is computed, the appropriate amount of vaporized liquid is defined as a source term for the given mesh cell, any heat transfer to a solid is stored as a convective term, and the net effect of heat transfer and mass addition to the gas is defined as a divergence. The mass and divergence terms are discussed in the sections that follow.

Equation (9.24) through Equation (9.27) are solved as a set of coupled implicit equations over the course of a gas phase time step. Details on this solution process are given in Appendix M.

#### 9.4.1 Filtered Volumetric Source Terms

The filtered volumetric source terms for mass and energy—which are required in the mass transport equation, Eq. (2.5), and the divergence expression, Eq. (2.10)—are obtained by summation of the individual particle source contributions with a given cell divided by the LES time step,  $\delta t_{\text{LES}}$ , and the local cell volume,  $V_c$ . The bulk mass and energy source terms are, respectively,

$$\dot{m}_{b,\alpha}''' = -\frac{\sum_p \sum_n \delta m_p^n}{\delta t_{\text{LES}} V_c} ; \quad \dot{q}_b''' = -\frac{\sum_p \sum_n \delta q_p^n}{\delta t_{\text{LES}} V_c}. \quad (9.35)$$

where  $n$  represents the sub-time step in the integration of the droplet mass and energy equations. The summation over  $p$  is over all the particles within the cell.

#### 9.4.2 Lagrangian Contribution to the Velocity Divergence

In practice, the filtered mass and energy source term contributions to the velocity divergence constraint are collected in a single term denoted `D_SOURCE` within the code,

$$D_{\text{SOURCE}} = \frac{1}{\rho} \sum_{\alpha} \frac{\bar{W}}{W_{\alpha}} \dot{m}_{b,\alpha}''' + \frac{1}{\rho c_p T} \left( \dot{q}_b''' - \sum_{\alpha} \dot{m}_{b,\alpha}''' \int_{T_b}^T c_{p,\alpha} dT' \right) \quad (9.36)$$

which is embedded in Eq. (3.26).

## 9.5 Fire Suppression by Water

The previous sections describe heat transfer from a liquid droplet to a gas, a solid, or both. Although there is some uncertainty in the values of the respective heat transfer coefficients, the fundamental physics are fairly well understood. However, when the droplets encounter burning surfaces, simple heat transfer correlations become more difficult to apply. The reason for this is that the water is not only cooling the surface and the surrounding gas, but it is also changing the pyrolysis rate of the fuel. If the surface of the fuel is planar, it

is possible to characterize the decrease in the pyrolysis rate as a function of the decrease in the total heat feedback to the surface. Unfortunately, most fuels of interest in fire applications are multi-component solids with complex geometry at scales unresolvable by the computational grid.

### 9.5.1 Droplet Transport on a Surface

When a liquid droplet hits a solid horizontal surface, it is assigned a random horizontal direction and moves at a fixed velocity until it reaches the edge, at which point it drops straight down at the same fixed velocity. This “dripping” velocity has been measured for water to be on the order of 0.5 m/s [104, 105]. While attached to a surface, the “droplet” is assumed to form a thin film of liquid that transfers heat to the solid, and heat and mass to the gas. The film thickness,  $\delta$ , is given by

$$\delta = \max \left( \delta_{\min}, \sum \frac{4}{3} \frac{\pi r_p^3}{A} \right) \quad (9.37)$$

where  $A$  is the area of the wall cell to which the droplet is attached. It is assumed that the minimum film thickness,  $\delta_{\min}$ , is  $1 \times 10^{-5}$  m. This prevents a very small amount of liquid from spreading across the entire cell width. It is also assumed that the liquid is opaque with regard to thermal radiation.

### 9.5.2 Reduction of Pyrolysis Rate due to Water

To date, most of the work in this area has been performed at Factory Mutual. An important paper on the subject is by Yu *et al.* [106]. The authors consider dozens of rack storage commodity fires of different geometries and water application rates, and characterize the suppression rates in terms of a few global parameters. Their analysis yields an expression for the total heat release rate from a rack storage fire after sprinkler activation

$$\dot{Q} = \dot{Q}_0 e^{-k(t-t_0)} \quad (9.38)$$

where  $\dot{Q}_0$  is the total heat release rate at the time of application  $t_0$ , and  $k$  is a fuel-dependent constant. This analysis is based on global water flow and burning rates. Equation (9.38) accounts for both the cooling of non-burning surfaces as well as the decrease in heat release rate of burning surfaces. In the FDS model, the cooling of unburned surfaces and the reduction in the heat release rate are computed locally. Thus, it is awkward to apply a global suppression rule. However, the exponential nature of suppression by water is observed both locally and globally, thus it is assumed that the local heat release rate per unit area can be expressed in the form [104, 105]

$$\dot{q}''(t) = \dot{q}_0''(t) e^{-\int k(t) dt} \quad (9.39)$$

where  $\dot{q}_0''(t)$  is the burning rate per unit area of the fuel when no water is applied and  $k(t)$  is a linear function of the local water mass per unit area,  $m_w''$ , expressed in units of kg/m<sup>2</sup>,

$$k(t) = a m_w''(t) \text{ s}^{-1} \quad (9.40)$$

Note that  $a$  is an empirical constant that is dependent on the material properties of the solid fuel and its geometrical configuration.

## 9.6 Using Lagrangian Particles to Model Complex Objects

There are many real objects that participate in a fire that cannot be modeled easily as solid obstructions that conform to the rectilinear mesh. For example, electrical cables, dry brush, tree branches, and so on, are potential fuels that cannot be well-represented as solid cubes, not only because the geometry is wrong,

but also because the solid restricts the movement of hot gases through the complex collection of objects. Additionally objects such as window screens also impose flow restrictions but are typically not resolvable in an engineering calculation. As a potential remedy for the problem, these objects can be modeled as discrete particles that are either spheres, cylinders or small sheets. Each particle can be assigned a surface type in much the same way as is done for solid obstructions that conform to the numerical grid. The particle is assumed to be thermally-thick, but for simplicity the heat conduction within the particle is assumed to be one-dimensional in either a cylindrical, spherical or cartesian coordinate system.

Particles interact with the radiation field through an additional sink term in the radiative transfer equation. For a grid cell with indices  $ijk$ , the integrated radiant intensity is reduced at rate

$$(-\nabla \cdot \dot{\mathbf{q}}_r'')_{ijk} = \sum \kappa_p (U_{ijk} - 4\sigma T_p^4) \quad (9.41)$$

where the summation is over all the particles within the cell. For individual intensity equations, see Eq. 6.42. The effective absorption coefficient for a single particle is given by

$$\kappa_p = \frac{\varepsilon_p A_p}{4 \delta x \delta y \delta z} \quad (9.42)$$

where  $\varepsilon_p$  is the emissivity of the particle,  $A_p$  is the surface area of the particle, and  $\delta x \delta y \delta z$  is the volume of the cell. The net radiative heat flux onto the surface of the particle is

$$\dot{q}_{r,p}'' = \varepsilon_p \left( \frac{U_{ijk}}{4} - \sigma T_p^4 \right) \quad (9.43)$$

### 9.6.1 Porous Media (Filters, Screens, Metal Meshes, and Similar Materials)

Air filters, screens, grating, and similar flow obstructions can all be considered as type of porous media. In general, material forming the porous media will have dimensions will below that of the grid size (e.g. 100 micron diameter filter fibers on a multi-cm grid). There is, therefore, no easy way to model these materials using solid obstructions. Lagrangian particles can; however, be used to represent both the drag and the mass of these materials. By placing particles in a plane or volume and assigning the particles a porous media drag law, the effects of the porous media can be modeled. The pressure drop over a length  $\Delta x$  through porous media is given by [107]:

$$\frac{\Delta p}{\Delta x} = \frac{\mu}{K} u + \rho \frac{Y}{\sqrt{K}} u^2 \quad (9.44)$$

where  $K$  is a permeability constant,  $Y$  is an inertial constant,  $u$  is the velocity normal to the screen,  $\rho$  is the density, and  $\mu$  is the viscosity of the gas. In the case of a non-isotropic media, the constants  $K$  and  $Y$  will vary with the flow direction. The force vector  $\mathbf{f}_b$  in Eq. (4.36) represents the momentum transferred from the screen to the gas:

$$\mathbf{f}_b = \left( \frac{\mu}{K} + \rho \frac{Y}{\sqrt{K}} |\mathbf{u}| \right) \mathbf{u} \quad (9.45)$$

For the special case of screens, gratings, and similar thin porous materials the permeability and inertial constants have been experimentally correlated to screen porosity.  $K$  and  $Y$  are functions of the screen porosity (free area/total area),  $\phi$ :[108]:

$$K = 3.44 \times 10^{-9} \phi^{1.6} \text{ m}^2 \quad ; \quad Y = 0.043 \phi^{2.13} \quad (9.46)$$

For a screen the force vector must account for the actual screen thickness,  $l$ , being less than that of the grid cell:

$$\mathbf{f}_b = l \left( \frac{\mu}{K} + \rho \frac{Y}{\sqrt{K}} |\mathbf{u}| \right) \left( \frac{u}{\delta x}, \frac{v}{\delta y}, \frac{w}{\delta z} \right) \quad (9.47)$$

This force term essentially spreads the pressure drop over the width of a grid cell.

## 9.7 Turbulent Dispersion

### 9.7.1 Massless Tracers

The effect of subgrid-scale turbulent fluid motion on the velocity and position of a Lagrangian particle may be accounted for using a random walk model [109]. The position of a tracer particle obeys the stochastic differential equation

$$dx^* = \left[ \tilde{\mathbf{u}} + \frac{1}{\bar{\rho}} \nabla(\bar{\rho} D_t) \right] dt + \sqrt{2D_t} d\mathbf{W} \quad (9.48)$$

where  $\mathbf{x}^*$  denotes the particle position (an asterisk signifies a particle property),  $\tilde{\mathbf{u}}$  is the resolved LES velocity,  $D_t$  is the turbulent diffusivity (taken from an eddy viscosity model, for example), and  $\mathbf{W}$  is an independent Wiener process. Notice that if no turbulent diffusion exists, the particle follows the resolved flow. The term added to the resolved velocity accounts for the deterministic mean drift and the random walk term (Wiener process) accounts for the reorientation effect of unresolved turbulent motion.

For those unfamiliar with stochastic differential equations, the Wiener process may be understood numerically as  $dW(t) = (\delta t)^{1/2} \zeta(t)$  in the limit  $\delta t \rightarrow 0$ , where  $\zeta(t)$  is an independent standardized Gaussian random variable [15]. In FDS,  $\zeta(t)$  are generated from a Box-Muller transform [110].

### 9.7.2 Massive Particles

For massive particles, a random walk model is not used. Instead, the fluid velocity used in the drag calculation is augmented with an isotropic velocity fluctuation taken from an estimate of the subgrid kinetic energy. Following [111], the gas velocity used in Eqs. (9.1) and (9.2) is given by

$$\mathbf{u} = \tilde{\mathbf{u}} + u' \boldsymbol{\zeta} \quad (9.49)$$

where  $\tilde{\mathbf{u}}$  is the resolved LES velocity,  $u' = \sqrt{\frac{2}{3} k_{sgs}}$ , and the components of  $\boldsymbol{\zeta}$  are independent standardized Gaussian random variables (zero mean and unit variance) generated from a Box-Muller transform [110]. The subgrid kinetic energy is estimated from the turbulent viscosity as  $k_{sgs} = (\mu_t / (\rho C_V \Delta))^2$ ;  $C_V = 0.1$  is the Deardorff eddy viscosity coefficient.



# Chapter 10

## Fire Detection Devices

FDS predicts the thermal environment resulting from a fire, but it relies on empirical models that describe the activation of various fire detection devices. These models are described in this section.

### 10.1 Sprinklers

The temperature of the sensing element (or “link”) of an automatic fire sprinkler is estimated from the differential equation put forth by Heskestad and Bill [112], with the addition of a term to account for the cooling of the link by water droplets in the gas stream from previously activated sprinklers

$$\frac{dT_l}{dt} = \frac{\sqrt{|\mathbf{u}|}}{\text{RTI}}(T_g - T_l) - \frac{C}{\text{RTI}}(T_l - T_m) - \frac{C_2}{\text{RTI}}\beta|\mathbf{u}| \quad (10.1)$$

where  $\mathbf{u}$  is the gas velocity, RTI is the response time index,  $T_l$  is the link temperature,  $T_g$  is the gas temperature in the neighborhood of the link,  $T_m$  is the temperature of the sprinkler mount (assumed ambient), and  $\beta$  is the volume fraction of (liquid) water in the gas stream. The sensitivity of the sprinkler link is characterized by its RTI value. The amount of heat conducted away from the link by the mount is indicated by the “C-Factor”,  $C$ . The RTI and C-Factor are determined experimentally. The constant  $C_2$  has been empirically determined by DiMarzo and co-workers [113, 114, 115] to be  $6 \times 10^6 \text{ K}/(\text{m/s})^{1/2}$ , and its value is relatively constant for different types of sprinklers.

The algorithm for heat detector activation is exactly the same as for sprinkler activation, except there is no accounting for conductive losses or droplet cooling. Note that neither the sprinkler nor heat detector models account for thermal radiation.

### 10.2 Heat Detectors

As far as FDS is concerned, a heat detector is just a sprinkler with no water spray. In other words, the activation of a heat detector is governed by Eq. (10.1), but with just the first term on the right hand side:

$$\frac{dT_l}{dt} = \frac{\sqrt{|\mathbf{u}|}}{\text{RTI}}(T_g - T_l) \quad (10.2)$$

Both the RTI and activation temperature are determined empirically.

## 10.3 Smoke Detectors

An informative discussion of the issues associated with smoke detection can be found in the SFPE Handbook chapter “Design of Detection Systems,” by Schifiliti, Meacham, and Custer [116]. The authors point out that the difficulty in modeling smoke detector activation stems from a number of issues: (1) the production and transport of smoke in the early stage of a fire are not well-understood, (2) detectors often use complex response algorithms rather than simple threshold or rate-of-change criteria, (3) detectors can be sensitive to smoke particle number density, size distribution, refractive index, composition, etc., and (4) most computer models, including FDS, do not provide detailed descriptions of the smoke besides its bulk transport. This last point is the most important. At best, in its present form, FDS can only calculate the velocity and smoke concentration of the ceiling jet flowing past the detector. Regardless of the detailed mechanism within the device, any activation model included within FDS can only account for the entry resistance of the smoke due to the geometry of the detector. Issues related to the effectiveness of ionization or photoelectric detectors cannot be addressed by FDS.

Consider the simple idealization of a “spot-type” smoke detector. A disk-shaped cover lined with a fine mesh screen forms the external housing of the device, which is usually mounted on the ceiling. Somewhere within the device is a relatively small sensing chamber where the smoke is actually detected in some way. A simple model of this device has been proposed by Heskstad [116]. He suggested that the mass fraction of smoke in the sensing chamber of the detector  $Y_c$  lags behind the mass fraction in the external free stream  $Y_e$  by a time period  $\delta t = L/u$ , where  $u$  is the free stream velocity and  $L$  is a length characteristic of the detector geometry. The change in the mass fraction of smoke in the sensing chamber can be found by solving the following equation:

$$\frac{dY_c}{dt} = \frac{Y_e(t) - Y_c(t)}{L/u} \quad (10.3)$$

The detector activates when  $Y_c$  rises above a detector-specific threshold.

A more detailed model of smoke detection involving two filling times rather than one has also been proposed. Smoke passing into the sensing chamber must first pass through the exterior housing, then it must pass through a series of baffles before arriving at the sensing chamber. There is a time lag associated with the passing of the smoke through the housing and also the entry of the smoke into the sensing chamber. Let  $\delta t_e$  be the characteristic filling time of the entire volume enclosed by the external housing. Let  $\delta t_c$  be the characteristic filling time of the sensing chamber. Cleary *et al.* [117] suggested that each characteristic filling time is a function of the free-stream velocity  $u$  outside the detector

$$\delta t_e = \alpha_e u^{\beta_e} \quad ; \quad \delta t_c = \alpha_c u^{\beta_c} \quad (10.4)$$

The  $\alpha$  and  $\beta$  parameters are empirical constants related to the specific detector geometry. Suggested values for these parameters are listed in the FDS User’s Guide [3]. The change in the mass fraction of smoke in the sensing chamber  $Y_c$  can be found by solving the following equation:

$$\frac{dY_c}{dt} = \frac{Y_e(t - \delta t_e) - Y_c(t)}{\delta t_c} \quad (10.5)$$

where  $Y_e$  is the mass fraction of smoke outside of the detector in the free-stream. A simple interpretation of the equation is that the concentration of the smoke that enters the sensing chamber at time  $t$  is that of the free-stream at time  $t - \delta t_e$ .

An analytical solution for Eq. (10.5) can be found, but it is more convenient to simply integrate it numerically as is done for sprinklers and heat detectors. Then, the predicted mass fraction of smoke in the sensing chamber  $Y_c(t)$  can be converted into an expression for the percent obscuration per unit length by computing:

$$\text{Obscuration} = \left( 1 - e^{-K_m \rho Y_c l} \right) \times 100 \% \text{ per length } l \quad (10.6)$$

where  $K_m$  is the mass extinction coefficient,  $\rho$  is the density of the external gases in the ceiling jet, and  $l$  is the unit of length over which the light is attenuated<sup>1</sup>. For most flaming fuels, a suggested value for  $K_m$  is  $8700 \text{ m}^2/\text{kg} \pm 1100 \text{ m}^2/\text{kg}$  at a wavelength of 633 nm [118].

The SFPE Handbook [116] has references to various studies on smoke detection and suggested values for the characteristic length  $L$ . FDS includes the one-parameter Heskestad model as a special case of the four-parameter Cleary model. For the Cleary model, the user must specify  $\alpha_e$ ,  $\beta_e$ ,  $\alpha_c$ , and  $\beta_c$ , whereas for the Heskestad model, only  $L = \alpha_c$  needs to be specified. Equation (10.5) is still used, with  $\alpha_e = 0$  and  $\beta_e = \beta_c = -1$ . Proponents of the four-parameter model claim that the two filling times are needed to better capture the behavior of detectors in a very slow free-stream ( $u < 0.5 \text{ m/s}$ ). Rather than declaring one model better than another, the algorithm included in FDS allows the user to pick these various parameters, and in doing so, pick whichever model the user feels is appropriate [119].

Additionally, FDS can model the behavior of beam and aspiration smoke detectors. For a beam detector, the user specifies the emitter and receiver positions and the total obscuration at which the detector will alarm. FDS will then integrate the obscuration over the path length using the predicted soot concentration in each grid cell along the path:

$$\text{Obscuration} = \left( 1 - e^{-K_m \int \rho Y_s dl} \right) \times 100 \% \quad (10.7)$$

where the integration is carried out over the path of the beam.

For an aspiration detector, the user specifies the sampling locations, the flow rate at each location, the transport time from each sampling point to the detector, the flow rate of any bypass flow, and the total obscuration at which the detector alarms. FDS computes the soot concentration at the detector by weighting the predicted soot concentrations at the sampling locations with their flow rates after applying the appropriate time delay. See the FDS User's Guide [3] for details.

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<sup>1</sup>Typically, the activation criterion for a spot-type smoke detector is listed as a percent obscuration per foot or per meter. For the former,  $l = 0.3048 \text{ m}$  and for the latter,  $l = 1 \text{ m}$ .



# Chapter 11

# Heating, Ventilation, and Air Conditioning (HVAC)

HVAC systems are found throughout the built environment. During a fire, HVAC ducts can serve as a path for heat and combustion products to be moved through a building, and the ducts can serve as a supply of fresh air for the fire. In some facilities, such as data centers and clean rooms, fire detection devices are placed inside of HVAC ducts. HVAC systems may also serve as part of the fire protection system for a building when used to exhaust smoke or maintain stairwell pressurization.

While simple defined velocity, mass flux, or pressure boundary conditions can represent inflow and outflow due to HVAC system, they cannot fully model an HVAC system. There is no coupling of the mass, momentum, and energy solutions amongst the multiple inlets and outlets comprising the HVAC network. To address this limitation, an HVAC network solver has been added to FDS.

The reader may find it useful to read a literature review on coupled hybrid modelling in fire applications [120] and review similar work in coupling CFD to a 1-D nodal model for analysis of ventilation in tunnel fires [121, 122].

## 11.1 Governing Equations

The overall HVAC solver is based on the MELCOR [89] thermal hydraulic solver. MELCOR is a computer program for simulating accidents in nuclear power plant containment buildings. The Fire and Smoke Simulator (FSSIM) [123], a network fire model, has shown prior success in using the MELCOR solver to model fire spread and smoke movement in the presence of complex ventilation systems, and the FDS implementation of the MELCOR solver is largely based on the implementation found in FSSIM. The coupling of the HVAC solver to the remainder of the FDS computation is in part based upon approaches taken in GOTHIC [124], another containment analysis code that couples CFD-like features for large containment volumes with a network model for piping and ventilation.

The MELCOR solver uses an explicit solver for the conservation equations of mass and energy combined with an implicit solver for the conservation of momentum equation. An HVAC system is represented as a network of nodes and ducts. A node represents where a duct joins with the FDS computational domain or where multiple ducts are joined such as at a tee joint. A duct segment in the network represents any continuous flow path not interrupted by a node, and as such may include multiple fittings (elbows, expansions, or contractions, etc.) and may have varying area over its length. The nodal conservation equations of mass, energy, and momentum (in that order) are:

$$\sum_j \rho_j u_j A_j = 0 \quad (11.1)$$

$$\sum_j \rho_j u_j A_j h_j = 0 \quad (11.2)$$

$$\rho_j L_j \frac{du_j}{dt} = (p_i - p_k) + (\rho g \Delta z)_j + \Delta p_j - \frac{1}{2} K_j \rho_j |u_j| u_j \quad (11.3)$$

where  $u$  is the duct velocity,  $A$  is the duct area, and  $h$  is the enthalpy of the fluid in the duct. The subscript  $j$  indicates a duct segment, the subscripts  $i$  and  $k$  indicate nodes (where one or more ducts join or where a duct terminates in a compartment).  $\Delta p$  is a fixed source of momentum (a fan or blower),  $L$  is the length of the duct segment, and  $K$  is the total dimensionless loss coefficient of the duct segment (which includes wall friction losses and minor losses).

Since nodes have no volume, the mass and energy conservation equations require that what flows into a node must also flow out. In the momentum equation the terms on the right hand side consist of the pressure gradient between the upstream and the downstream node, the buoyancy head, pressure rise due to an external source (e.g., a fan or blower), and the pressure losses due to wall friction or the presence of duct fittings. In its default mode of operation, the transport delay of mass and energy along the length of a duct is not accounted for. An optional model for this delay is available and described in Sec. 11.6.

## 11.2 Solution Procedure

The momentum equation (11.3) is non-linear with respect to velocity due to the loss term. Additionally, the pressure difference between two nodes in the network is impacted by the pressure change at all nodes coupled to that duct either directly (part of the same duct network) or indirectly (connected to the same compartment as another duct network). Solving the momentum equation, requires accounting for both of these. This is done with the following discretization:

$$u_j^{n+1} = u_j^n + \frac{\Delta t^n}{\rho_j L_j} \left[ (\tilde{p}_i^n - \tilde{p}_k^n) + (\rho g \Delta z)_j^n + \Delta P_j^n - \frac{1}{2} K_j \rho_j \left( |u_j^{n-} + u_j^{n+}| u_j^{n+} - |u_j^{n+} | u_j^{n-} \right) \right] \quad (11.4)$$

The superscripts  $n+$  and  $n-$  on the velocity are used to linearize the flow loss in a duct to avoid a non-linear differential equation for velocity. The  $n+$  superscript is the prior iteration value and the  $n-$  is either the prior iteration value or zero if flow reversal occurred between iterations. This approach is used to speed convergence when duct flows are near zero to avoid large changes in  $K$  if the forward and reverse losses are markedly different. Note that the node pressures are not expressed as  $P_i^n$ , but rather as  $\tilde{p}_i^n$ . This indicates an extrapolated pressure at the end of the current time step rather than the actual pressure at the end of the time step. The pressure in a compartment is a function of the mass and energy flowing in and out. If that compartment is connected to other compartments by doors or other openings, then the pressure is also dependent upon flows into and out those other compartments. Those mass and energy flows include both those being predicted by the HVAC model and those being predicted by the CFD model. For example, in Fig. 11.1, the un-shaded compartments have pressure solutions that are dependent upon the flows predicted by both the HVAC model and the CFD model and all of those compartments need to be included in the extrapolated pressure for those compartments. Since the two models are not fully coupled, the extrapolated pressure is an estimate of the pressure at the end of the time step based upon the pressure rise for the prior time step.

The extrapolated pressure for a compartment can be determined by using Eq. (3.31) and correcting the integral over velocity for the current solution of all interdependent HVAC flows into or out of an FDS pressure zone:

$$\tilde{p}_i^n = p_i^{n-1} + \left( \frac{dp_i^{n-1}}{dt} + \frac{\sum_j u_j^n A_j^{n-1} - \sum_j u_j^{n+1} A_j^n}{\int_{\Omega_m} P dV} \right) \Delta t^n = \tilde{p}_i^{*n} - \frac{\Delta t^n \sum_j u_j^{n+1} A_j^n}{\int_{\Omega_m} P dV} \quad (11.5)$$

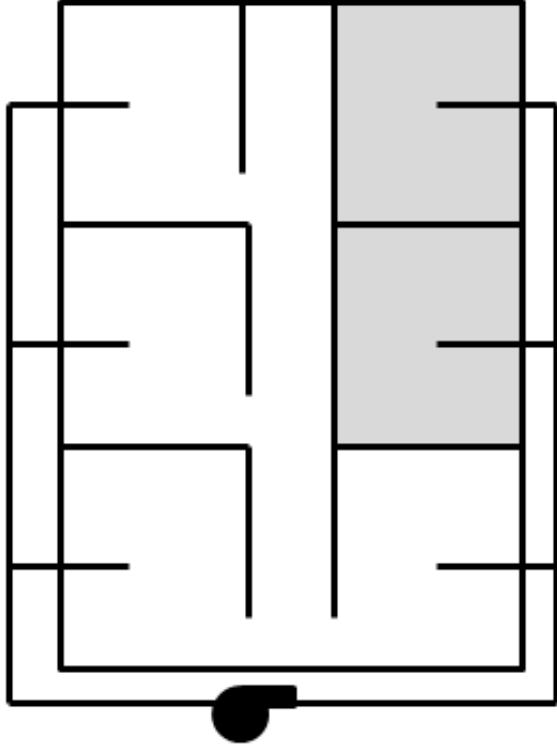


Figure 11.1: Illustration of interdependent pressure solutions. All unshaded compartments have pressures that are dependent upon each other.

If the summation term for the velocities being predicted in this time step is removed from Eq. (11.5) and placed on the left hand side of Eq. (11.4) and the remaining terms of Eq. (11.5) are placed on the right hand side we obtain the following:

$$u_j^{n+1} \left( 1 + \frac{\Delta t^n K_j}{2L_j} |u_j^{n-} + u_j^{n+}| \right) - \frac{\Delta t^{n2}}{\rho_j L_j} \frac{\sum_{j \in i} u_j^{n+1} A_j^n - \sum_{j \in k} u_j^{n+1} A_j^n}{\int_{\Omega_m} P \, dV} = \\ u_j^n + \frac{\Delta t^n}{\rho_j L_j} (\tilde{p}_i^{*n} - \tilde{p}_k^{*n} + (\rho g \Delta z)_j + \Delta p_j) + \frac{\Delta t^n K_j}{2L_j} |u_j^{n+}| |u_j^{n-}| \quad (11.6)$$

If node  $i$  or node  $k$  for duct  $j$  in Eq. (11.6) is an internal duct node, then extrapolated pressures are not computed and the actual node pressure is solved for. Applying Eq. (11.6) to each duct results in a linear set of equations. Adding additional equations to the set for the mass conservation at internal duct nodes, results in complete set of equations. The solution scheme is as follows:

1. Determine the boundary conditions at all points where the HVAC network joins the FDS computational domain using the previous time step values.
2. Compute the extrapolated pressures for each pressure zone using the previous iteration (previous time step if the first iteration).
3. Assemble the linear set of equations for conservation of momentum and conservation of mass.
4. Solve the equation and check the solution for errors in mass conservation, flow reversal over the time step, and the magnitude of change in the velocity solution for each duct. If any convergence check fails,

the solution is re-iterated with new extrapolated pressures. Density and enthalpy values are taken as the upwind values in each iteration. After each iteration, the temperature and density of each node are updated using the velocity and pressure solution. The node temperature is computed by summing the enthalpy flows into the node and computing the average temperature that represents the total enthalpy. Density is then updated using the equation of state and the new temperature.

### 11.2.1 Filtration

Filters have two effects on the flow in an HVAC network. First, a filter causes a flow loss whose magnitude is a function of the mass loading of the filter. Second, a filter removes one or more species from flow going through the filter. The removal rate is a function of the filtration efficiency for that species and its massflow rate through the filter. Filter losses are evaluated using the filter loading at the start of each time step. This loss is applied to the upstream duct. The filter loss is computed as a function of the total filter loading using either a linear ramp or a user defined table. The total loading of the filter is determined by summing the mass of each species trapped times a weighting factor for that species.

The filter is assumed to remove a fixed fraction (the filter's efficiency) of the species being trapped by the filter. Each species can be given its own removal efficiency. Eq. (11.1) for a filter is therefore given as:

$$u_{\text{out}} \rho_{\text{out}} A_{\text{out}} = u_{\text{in}} \rho_{\text{in}} A_{\text{in}} - \sum_{\alpha} u_{\text{in}} \rho_{\text{in}} A_{\text{in}} Y_{\alpha,\text{in}} E_{\alpha} = u_{\text{in}} \rho_{\text{in}} A_{\text{in}} \left( 1 - \sum_{\alpha} Y_{\alpha,\text{in}} E_{\alpha} \right) \quad (11.7)$$

where  $\alpha$  is a species being filtered and  $E_{\alpha}$  is its removal efficiency.

### 11.2.2 Node Losses

Nodes in the HVAC systems can include inlet and outlets, tees, and other duct fittings. Each of these impose a flow loss ; however, as seen in Eq. (11.6), flow loss terms appear only in the equations for a duct. This means losses that are physically associated with a node must be expressed numerically as equivalent losses in the ducts attached to the node. The losses also need to be applied in a manner that represents the flow conditions within the node. For example, if a tee has flow into one leg and out of two legs, it would not make sense to apply the loss to the upstream leg as there would be no way to distinguish losses due to any changes of the flow splitting in the downstream legs. Loss terms are applied as follows:

1. If there is no flow at the node, then each duct connected to the node is assigned the average of all the losses for flows to the duct from all other ducts.
2. If there is flow into only one connected duct, then each outflowing duct is assigned the flow loss for flow from the inlet duct to that specific outflowing duct.
3. If there is flow out of only one connected duct, then each inflowing duct is assigned the flow loss for flow from that inlet duct to the outlet duct corrected for any change in duct area from inlet to outlet (node losses are input as a function of the downstream duct area).
4. If there is flow into multiple ducts and out of multiple ducts, then each outgoing duct is given the average loss from the inflowing ducts weighted by the volume flow through those ducts.

### 11.2.3 Duct Losses

The total flow loss in the duct,  $K_j$ , is the sum of fitting losses in the duct (e.g. elbows, expansion/reduction, orifice plates),  $K_{\text{minor}}$ , plus losses due to wall friction,  $K_{\text{wall}}$ . Minor losses are a user input. If a duct roughness

is set, wall friction losses are modeled as follows:

$$K_{\text{wall}} = \frac{fL}{D} \quad (11.8)$$

where  $D$  is the duct diameter.  $f$  is determined from the Colebrook equation. However, since this equation does not have an analytical solution, an approximation by Zigrang and Sylvester is used [125].

$$\frac{1}{\sqrt{f}} = -2 \log_{10} \left( \frac{\epsilon/D}{3.7} - \frac{4.518}{\text{Re}_D} \log_{10} \left( \frac{6.9}{\text{Re}_D} + \left( \frac{\epsilon/D}{3.7} \right)^{1.11} \right) \right) \quad (11.9)$$

where  $\epsilon$  is the absolute roughness of the duct.

#### 11.2.4 Heating and Cooling Coils

A duct can contain a heating or cooling coil. These either add or remove heat from the mass flowing in a duct. This enthalpy change is then added to the duct enthalpy flow at the downstream node prior to computing the node temperature. Two models are available. The first model simply adds or removes heat at a specified rate as long as the coil is operating. The second model uses a heat exchanger effectiveness model in which four parameters are specified: the specific heat of the working fluid ( $c_{p,\text{fl}}$ ), the temperature of the working fluid, ( $T_{\text{fl}}$ ), the mass flow rate of the working fluid ( $\dot{m}_{\text{fl}}$ ), and the effectiveness ( $\eta$ ). The coil heat flow rate,  $\dot{q}_{\text{coil}}$ , is then computed as follows:

$$T_{\text{out}} = \frac{\dot{m}_{\text{duct}} c_{p,\text{duct,in}} T_{\text{duct,in}} + \dot{m}_{\text{fl}} c_{p,\text{fl}} T_{\text{fl}}}{\dot{m}_{\text{duct}} c_{p,\text{duct,in}} + \dot{m}_{\text{fl}} c_{p,\text{fl}}} \quad (11.10)$$

$$\dot{q}_{\text{coil}} = \dot{m}_{\text{fl}} c_{p,\text{fl}} (T_{\text{fl}} - T_{\text{out}}) \eta \quad (11.11)$$

### 11.3 Leakage

With rare exceptions, walls, floors, and ceilings are not air tight. Gaps around windows and doors and openings for electrical, mechanical, and other systems provide small flow paths through obstructions. These flow paths can be modeled as an equivalent HVAC system where each leakage path is a single duct. Two approaches are available for modeling leakage.

In the first approach one or more surfaces in the domain are defined as being part of a leakage path between two pressure zones. An HVAC duct is created whose area is the leakage area and whose nodes are all wall cells belong to the surfaces that make up the leakage path. In this approach only the background pressure,  $\bar{p}_m$ , in each zone is used. of the duct is total leakage area and the terminal nodes of the duct can be considered the entire area of the surfaces defined as participating in that flow path. Since only the background pressure is used, this approach can only be used for leakage between pressure zones.

The second approach, a specific area is identified as the inlet and outlet for the leakage path. This approach is useful to represent small openings in known locations that are subgrid such as a the undercut beneath a typical interior door. Similar to the first approach an HVAC dut is created whose area is the leakage area and whose nodes are the wall cells belonging to the identified inlet and outlet regions. This approach uses the total pressure; therefore, it can be applied in cases where the inlet and outlet are in the same pressure zone.

For more details see the FDS User's Guide [3].

## 11.4 Coupling the HVAC solver to FDS

### 11.4.1 Boundary Conditions for the HVAC Solver

Prior to updating the HVAC solution, the inlet conditions at each duct node are determined by summing the mass and energy of the gas cells next to duct node and averaging the pressure. The total mass and energy along with the average pressure are then used to determine the average temperature.

$$\bar{\rho}_i = \frac{\sum_j \rho_j A_j}{\sum_j A_j} ; \quad \bar{Y}_{\alpha,i} = \frac{\sum_j Y_{\alpha,j} \rho_j A_j}{\sum_j \rho_j A_j} ; \quad \bar{P}_i = \frac{\sum_j P_j A_j}{\sum_j A_j} \quad (11.12)$$

$$\bar{h}_i = \frac{\sum_j \rho_j A_j c_p(T_j, Y_j)}{\sum_j \rho_j A_j} ; \quad \bar{T}_i = \frac{\bar{h}_i}{c_p(\bar{T}_i, \bar{Y}_i)} \quad (11.13)$$

where  $i$  is a duct node and  $j$  are the gas cells adjacent to the node.

### 11.4.2 Boundary Conditions for the FDS Hydrodynamic Solver

For wall cells containing inflow from an HVAC duct that is not leakage flow, the surface temperature,  $T_w$ , is set to the value in the connected duct. If the flow is a leakage flow, then  $T_w$  is computed based on the thermal properties assigned to the surface (see Chapter 7). The remaining wall boundary conditions are computed as follows:

$$\dot{m}''_{\alpha} = Y_{\alpha,d} \dot{m}'' ; \quad \dot{m}'' = \frac{u_d \rho_d A_d}{A_v} \quad (11.14)$$

where the subscript  $d$  is the attached duct and  $A_v$  is the total area of the vent (which in the case of leakage flow is the total area of all surfaces for that leak path).

$$u_w = \frac{\dot{m}''}{\rho_w} ; \quad \rho_w = \frac{p \bar{W}}{R T_w} \quad (11.15)$$

$$Y_{\alpha,w} = \frac{\dot{m}''_{\alpha} + \frac{2\rho_w D Y_{\alpha,gas}}{\delta_n}}{\frac{2\rho_w D}{\delta_n} + u_w \rho_w} \quad (11.16)$$

The above three equations are solved iteratively with a limit of 20 iterations (typically only one or two iterations are needed).

For wall cells with outflow to an HVAC duct, the wall boundary conditions are set to gas cell values except for a leakage flow where the temperature is computed based on the thermal properties assigned to the surface.

## 11.5 Fan Curves

The flow rate induced by a fan is a function of the pressure across the fan in the duct run containing the fan (more back pressure for a fan results in less flow). The pressure in the duct run is itself a function of the flow through the fan (higher velocity results in a larger duct pressure drop). A plot of these two functions are respectively called the fan curve and the system curve. For a given set of conditions, the fan flow will be the intersection point of these two curves also called the operation point, see Fig. 11.2.

When a fan curve is specified for an FDS simulation, FDS must determine the correct fan pressure to use ( $\Delta p_f$  in Eq. 11.3). Two approaches for this are available in FDS.

The first is a rapid estimation approach that relies upon typical building construction and HVAC system design. Generally in a building, large pressure differences do not exist between rooms. It is also generally the case that HVAC systems have only a single fan in any duct run. In this case the fan pressure for a given time step can be estimated as the pressure corresponding to the current flow rate in the duct containing the fan.

This approach cannot be relied upon if there are multiple fans in a duct run as the approach will not correctly account for the combined effect. This approach can also not be relied upon if large pressure differences occur across the endpoints of the duct run. This might occur with a rapidly growing fire in a relatively well sealed space. In these cases one needs to define a system curve and find its intersection with the fan curve. This is the second approach in FDS.

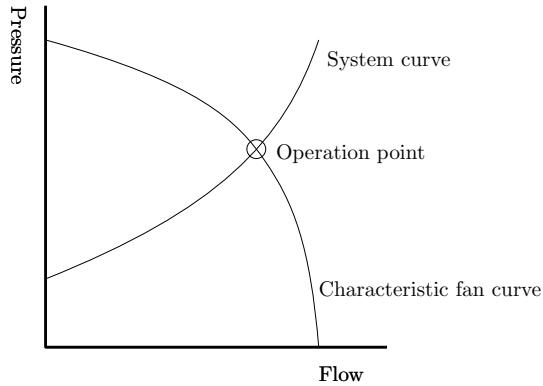


Figure 11.2: Illustration of the fan curve, system curve, and operation point for an HVAC fan.

A challenge in fire simulation is that the system curve is a function of the pressures seen at the end points of a duct run and the flow rate through the duct run. With a fire whose size changes over time, the pressures will also change. Therefore, a pre-calculated system curve cannot be used. Instead an alternate method [126] is used. This method relies on the observation that pressure and flow in an HVAC system have a quadratic relationship. Pressure drop in a duct is a function of the duct velocity squared. The following steps are taken at each time step to find the operation point for each fan:

1. Locate all duct runs that contain fans. A duct run is any collection of directly interconnected HVAC components. See Fig. 11.3 for two examples. The figure shows two sets of two compartments connected by HVAC systems. In each set, each compartment has a supply and exhaust duct. On the left side, the two exhaust ducts connect to a duct containing a fan which splits into a supply duct for each compartment. In this case there is one duct run. Every HVAC component is connected to every other HVAC component by a path containing only HVAC components. On the right side, the exhaust for each compartment is connected to its own fan which becomes the supply for the other compartment. In this case there are two duct runs. There is no direct path through only HVAC components that join the two ducts.
2. Determine the current pressures at each inlet and outlet for each duct run containing a fan.
3. For each fan, using just the duct run containing the fan, solve for the flow in the duct run given the current endpoint pressures.
4. For each fan, using just the duct run containing the fan, solve for the flow in the duct run given the current endpoint pressures and assuming the fan is applying its maximum pressure to the duct run.

5. For each fan, fit a quadratic between the two system points for that fan.
6. Use Newton's method to find the intersection between the user specified fan curve and the system curve. The fan pressure,  $\Delta p_j$ , for the time step is set to the pressure at the intersection. If there is no intersection, the fan pressure is set to either the maximum pressure for the fan or zero pressure. It is set to zero pressure if the system curve flow for zero fan pressure exceeds the maximum flow defined in the fan curve. It is set to maximum pressure if there is reverse flow in the system curve at maximum pressure.

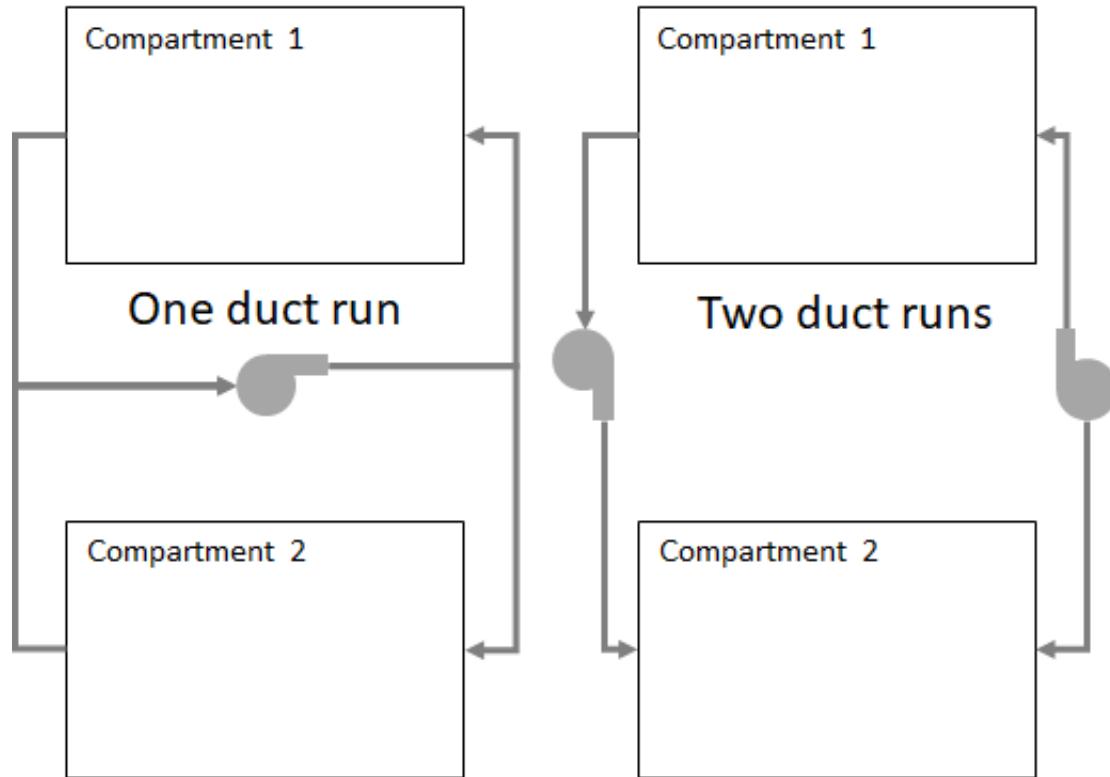


Figure 11.3: Illustration determining duct runs. Left side has a single duct run. Right side has two duct runs.

## 11.6 Duct Transport Delay

In its default mode of operation, the HVAC solver does not account for mass storage in ducts. That is over a time step any mass and energy entering a duct, leaves the duct. An optional submodel can be invoked to account for this transport delay. This submodel divides a duct into cells of equal length  $\Delta x$  and solves a one dimensional advection equations for mass and energy transport in the duct [127]. In the discretization, temperature, specific heat, species mass fractions, and density are stored at cell centers. The velocity in duct is taken as the value obtained from the solution of Eq. (11.1) through Eq. (11.3). An explicit Runge-Kutta method is used with a Godunov upwinding scheme. Diffusive transport is ignored as this is generally a minor effect in an HVAC system. The species mass and energy conservation equations for updating cell  $c$  in duct  $j$  from timestep  $n$  to  $n + 1$  are:

$$(\rho Y_\alpha)_c^{n+1} = \rho_j u_j^n - \frac{\Delta t}{\Delta x} (\rho u)_j^n \left( Y_{\alpha,c}^n - Y_{\alpha,c-1}^n \right) \quad (11.17)$$

$$(\rho h)_c^{n+1} = \rho_j u_j^n - \frac{\Delta t}{\Delta x} (\rho u)_j^n \left( h_c^n - h_{c-1}^n \right) \quad (11.18)$$

Updating duct conditions begins at the upwind duct node and marches over the duct length to the downwind duct node. Cell temperatures are extracted from the cell enthalpy  $h_c$ . This solution is done over groups of ducts that form a duct network basis. A duct network is a set of ducts and nodes that are interconnected via a flow path through HVAC components. The time step  $\Delta t$  is taken as the minimum of the current simulation time step and the minimum value of  $\frac{\Delta x_j}{2u_j}$  over all ducts in the network. In the summation terms in Eq. (11.1) and Eq. (11.2), the mass flow and enthalpy flow into a node from a discretized duct are obtained by integrating those flows using the values in the last cell over each time step taken.



# Chapter 12

## Unstructured Geometry (Beta)

Unstructured geometry is treated using a “cut-cell” (CC) method, following the seminal work of Michael Aftosmis and Marsha Berger [128, 129, 130]. The term cut-cell refers to the gas phase region of a Cartesian grid cell that has been “cut” by the unstructured geometry partitioning the cell into gas and solid sub-volumes. Faces of the cut-cell that have also been carved from cartesian faces and connected to other cut-cells are called *gasphase cut-faces* (these are gas-gas interfaces). The faces on the boundary of the solid in a cut-cell are *in-boundary cut-faces* (these are gas-solid interfaces). There are also faces of a cut-cell which are not cut and which connect the cell to a neighboring regular cartesian gas phase cell; these faces are referred to as *regular faces* (also a gas-gas interface).

The introduction of cut-cells surrounding an embedded boundary generates the need for an unstructured solution algorithm. In particular, the pressure equation is unstructured and so fast trigonometric solvers are not viable (i.e. the FFT Poisson solver). Instead, an unstructured local matrix (ULMAT) solver has been developed using the Pardiso and Sparse Cluster Solvers available in the Intel Math Kernel Library (MKL).

The following sections describe the finite volume methods used to discretize the governing equations in the unstructured cut-cells surrounding embedded boundaries.

### 12.1 Scalar Transport Discretization for Complex geometry

#### 12.1.1 Multispecies Mass Balance Equations

Consider a set of gaseous, reacting chemical species  $\alpha = 1, \dots, N$  flowing on a given spatial domain  $\Omega \in \mathbb{R}^n$ ,  $n = 2, 3$ , with boundary  $\partial\Omega$ , parameterized by an Eulerian reference frame  $N$ . These species are transported on a given point  $\mathbf{x}$  in space with velocity  $\mathbf{u}_\alpha(\mathbf{x}, t)$  respect to  $N$ , and a mass weighted average velocity  $\mathbf{u}(\mathbf{x}, t)$

$$\mathbf{u} = \frac{\sum_{\alpha=1}^N \rho_\alpha \mathbf{u}_\alpha}{\rho}, \quad \rho = \sum_{\alpha=1}^N \rho_\alpha \quad (12.1)$$

where space and time dependencies are not shown for simplicity,  $\rho_\alpha(\mathbf{x}, t) = \rho(\mathbf{x}, t)Y_\alpha(\mathbf{x}, t)$ ,  $\rho$  is the mixture density and  $Y_\alpha = \rho_\alpha/\rho$  is species  $\alpha$  mass fraction. Definition:  $\rho_\alpha(\mathbf{x}, t)$  is the amount of mass of species  $\alpha$  on a given differential volume  $d\Omega$  centered in  $\mathbf{x}$ , while  $\rho(\mathbf{x}, t)$  is the mass of *all* species in  $d\Omega$ .

As defined, each species  $\alpha$  has a velocity  $\mathbf{u}_\alpha(\mathbf{x}, t)$  which is different from the average mixture velocity  $\mathbf{u}(\mathbf{x}, t)$ . This difference quantity is called the *diffusion velocity* of species  $\alpha$ ,  $\mathbf{V}_\alpha$

$$\mathbf{V}_\alpha(\mathbf{x}, t) = \mathbf{u}_\alpha(\mathbf{x}, t) - \mathbf{u}(\mathbf{x}, t) \quad (12.2)$$

The transport-reaction mass balance equation for each individual species  $\alpha$  is given by

$$\frac{\partial \rho_\alpha}{\partial t} + \nabla \cdot (\rho_\alpha \mathbf{u}_\alpha) = \dot{m}'_\alpha''' , \quad \alpha = 1, \dots, N \quad (12.3)$$

in  $\Omega$ , where  $\dot{m}'_\alpha'''(\mathbf{x}, t)$  is the chemical reaction volume source (area source for  $n = 2$ , sink if negative) of species  $\alpha$ . The problem is completely defined stating initial and boundary conditions for  $\rho_\alpha(\mathbf{x}, t)$  as well as time dependent fields  $\mathbf{u}_\alpha(\mathbf{x}, t)$ ,  $\dot{m}'_\alpha'''(\mathbf{x}, t)$ .

Noting that  $\rho_\alpha = \rho Y_\alpha$ , and using equation (12.2) in the previous

$$\frac{\partial \rho Y_\alpha}{\partial t} + \nabla \cdot (\rho Y_\alpha (\mathbf{u} + \mathbf{V}_\alpha)) = \dot{m}'_\alpha''' , \quad \alpha = 1, \dots, N \quad (12.4)$$

We see from these equations that the evolution of these species can be considered as a function of the mixture  $\rho$  and average  $\mathbf{u}$ , and written as such, depending on the other species through Eqs. (12.1), and through the mass conservation statement on reactive sources

$$\sum_{\alpha=1}^N \dot{m}'_\alpha'''(\mathbf{x}, t) = 0 \quad (12.5)$$

In Eq. (12.4), the following convective and diffusive fluxes per unit time and area are found:

$$\mathbf{J}_{c\alpha} = \rho Y_\alpha \mathbf{u} \quad (12.6)$$

$$\mathbf{J}_{d\alpha} = \rho Y_\alpha \mathbf{V}_\alpha \quad (12.7)$$

The definition of diffusion velocities  $\mathbf{V}_\alpha$  is a central component of multispecies mass transport. For atmospheric combustion we consider the binary diffusion of all species respect to Nitrogen, the most abundant (*background*) species. This simplification allows us to uncouple the diffusive fluxes, using Ficks Law of diffusion

$$\mathbf{J}_{d\alpha} = \rho Y_\alpha \mathbf{V}_\alpha = -\rho D_\alpha \nabla Y_\alpha \quad (12.8)$$

where  $D_\alpha$  is the binary diffusivity coefficient respect to the background species.

Using this last expression in Eq. (12.4) we arrive to the basic form of the species balance equations based on mass fractions, and used in FDS

$$\frac{\partial \rho Y_\alpha}{\partial t} + \nabla \cdot (\rho Y_\alpha \mathbf{u}) = \nabla \cdot (\rho D_\alpha \nabla Y_\alpha) + \dot{m}'_\alpha''' , \quad \alpha = 1, \dots, N \quad (12.9)$$

As will be seen in next section, a spatially discretized version of the previous equations in a domain divided in  $ii = 1, \dots, n_{tot}$  computational cells is given by:

$$\frac{\partial (\rho Y_\alpha)_{ii}}{\partial t} V_{ii} = F_{\alpha,ii}^{adv} + F_{\alpha,ii}^{diff} + \dot{m}_{\alpha,ii}'' + F_{\alpha,ii}^{BC} , \quad \alpha = 1, \dots, N \quad (12.10)$$

where  $V_{ii}$  is the cell volume,  $(\rho Y_\alpha)_{ii}$  are the cell unknown partial densities,  $F_{\alpha,ii}^{adv}$ ,  $F_{\alpha,ii}^{diff}$ ,  $\dot{m}_{\alpha,ii}''$ ,  $F_{\alpha,ii}^{BC}$  are terms due to advection, diffusion, reaction and boundary conditions. Given that the time integration method we use is explicit, these vectors make use of available values of  $(\rho Y_\alpha)_{ii}$  from a previous time step or substep.

### 12.1.2 Finite Volume Method on Staggered Grid in 2D

The finite volume discretization (FV) starts by considering the integral form of Eq. (12.9) over a cell control volume  $\Omega_{ii}$ . For a cut-cell  $ii$  individualized by the index pair  $(i, j)$  we have

$$\int_{\Omega_{ii}} \frac{\partial \rho Y_\alpha}{\partial t} d\Omega + \int_{\Omega_{ii}} \nabla \cdot (\rho Y_\alpha \mathbf{u}) d\Omega = - \int_{\Omega_{ii}} \nabla \cdot (\mathbf{J}_{d\alpha}) d\Omega + \int_{\Omega_{ii}} \dot{m}'' d\Omega \quad (12.11)$$

Assuming a time independent control volume, the time derivative and source terms are approximated by

$$\int_{\Omega_{ii}} \frac{\partial \rho Y_\alpha}{\partial t} d\Omega = \frac{\partial}{\partial t} \int_{\Omega_{ii}} \rho Y_\alpha d\Omega = \frac{\partial \overline{\rho Y_{\alpha i,j}}}{\partial t} V_{i,j} \quad (12.12)$$

$$\int_{\Omega_{ii}} \dot{m}'' d\Omega = \overline{\dot{m}''_{\alpha i,j}} V_{i,j} \quad (12.13)$$

where  $V_{i,j}$  is the volume of cell  $(i, j)$ . These cell averaged quantities (denoted by tildes) match the cell centroid values of the corresponding scalar fields up to second order spatial accuracy. A consequence of this known fact is that using a second order finite difference method (FD), or a *difference* finite volume approach with same definition for diffusion flux spatial derivatives and interpolation on the advective term, will lead to identical discretization matrices (they will vary up to a volume factor) on uniform cartesian grids. The difference between these two numerical discretization methods arises from the source  $\dot{m}'''$ , which for a given cell is the cell centroid value  $\dot{m}'''_{\alpha(i,j)}$  in FD, and it is the cell average  $\widetilde{\dot{m}'''_{\alpha i,j}}$  in FV. To discretize both diffusive and advective terms in equation (12.11), we make use of the divergence theorem. In the following we drop the tildes to simplify the notation, keeping in mind that using FV, quantities will always be cell or face averaged where it corresponds.

### 12.1.3 Finite-Volume Discretization in Cut-Cells

The treatment of irregular gas phase Eulerian grid faces and cells that remain after the inclusion of immersed bodies is described in this section. We assume (see Fig. 12.1a) the cut-cell definition algorithm has been successful in defining the following:

- All GASPHASE faces in the  $x$  and  $y$  directions, which are tagged as regular or cut-faces. For these, we also know vertex points, areas (length in 2D), and face centroid location. We know for faces that are GASPHASE cut-faces the  $i, j$  coordinates of the face they belong to.
- All INBOUNDARY faces, which arise from the intersection of the immersed bodies surface elements and the Eulerian grid cells. For these we know vertex points, areas (length in 2D), and face centroid location.
- All GASPHASE cut cells and regular cells. We know which faces define their boundary. For cut-cells, a list of cut-faces (type GASPHASE and INBOUNDARY) is provided. Also we know their volume and face centroid location.

### 12.1.4 Cut-Cell Discretization of Diffusive Term

Consider the FV discretization of the diffusive term (equation (12.11)) on cut-cell  $ii$  in Fig. 12.1b. We have

$$\int_{\Omega_{ii}} \nabla \cdot (\mathbf{J}_{d\alpha}) d\Omega = \int_{\partial\Omega_{ii}} (-\rho D_\alpha \nabla Y_\alpha) \cdot \hat{\mathbf{n}}_{ii} d\partial\Omega = \sum_{k=1}^{nf_c=6} (-\rho D_\alpha \nabla Y_\alpha)_k \cdot \hat{\mathbf{n}}_{ii,k} A_k \quad (12.14)$$

The  $nf_c = 6$  faces that compose the boundary of the cut-cell can be divided in:

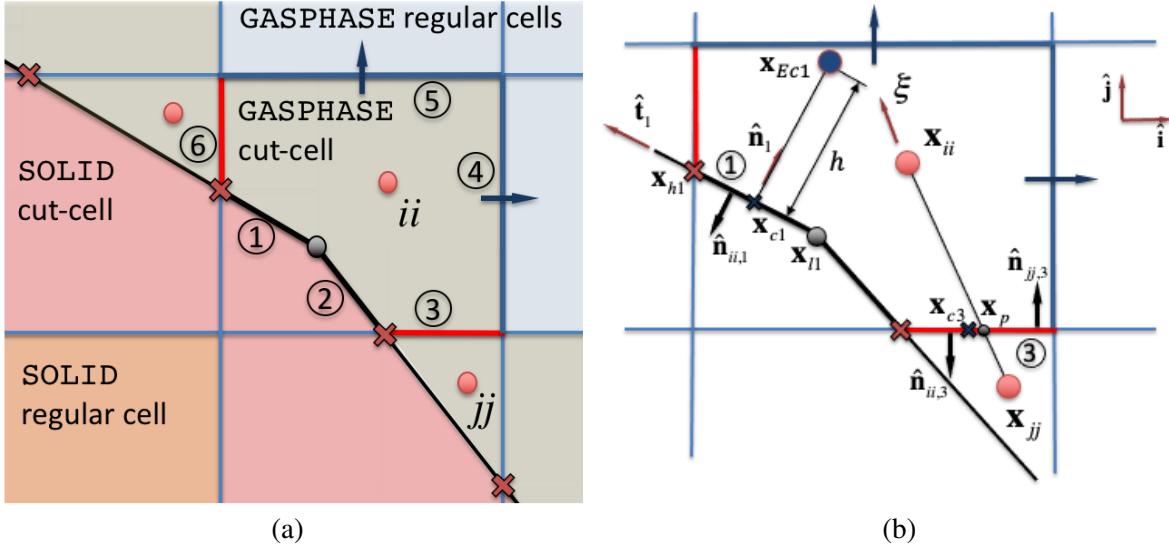


Figure 12.1: Cut-cell FV discretization in 2D: (a) GASPHASE Cut-cell  $ii$  is bounded by: **1, 2** INBOUNDARY faces, **3, 6** GASPHASE cut-faces, and **4, 5** GASPHASE regular faces. (b) Geometrical elements used in the discretization of the diffusive term for GASPHASE cut-face **3**, and INBOUNDARY face **1**.

### A. Faces $k = 4, 5$ are **regular GASPHASE Faces Connecting a Cut-Cell with a Regular Cell**:

The treatment of these is as done in regular cartesian faces, with the caveat that the diffusive flux computation now involves the cut-cell centroid location (i.e. to compute spatial derivatives of  $Y_\alpha$  and interpolation of  $\rho D_\alpha$  in equation (12.14)).

### B. Faces $k = 3, 6$ are **GASPHASE Cut-Faces**:

In order to compute the discrete term of Eq. (12.14) on these faces, the factor  $\rho D_\alpha$  needs to be interpolated from cell centroids to the face centroid and the spatial derivative of  $Y_\alpha$  computed at the face centroid. The use of face centroids is required to maintain spatial accuracy when going from the left to right hand side of Eq. (12.14) in difference FV methods. To maintain accuracy of the overall cut cell method, interpolation and differentiation to the cut-face centroid must also be sufficiently accurate.

The following discussion is fairly general, in the sense that it is agnostic to the interpolation methods employed (Lagrange polynomials, isoparametric, least squares, etc.). Consider a stencil of points  $e = 1, \dots, ne$ , where the scalar  $Y_\alpha$  is assumed defined (i.e. cell centroids). Then for a location of interest  $ck$  (i.e. the centroid of GASPHASE cut-face  $k = 3, 6$ ), interpolation and derivatives of  $Y_\alpha$  at  $ck$  can be obtained as

$$Y_\alpha(\mathbf{x}_{ck}) = \sum_{e=1}^{ne} \phi_e(\mathbf{x}_{ck} - \mathbf{x}_e) Y_{\alpha e} \quad (12.15)$$

$$\frac{\partial Y_\alpha(\mathbf{x}_{ck})}{\partial x_i} = \sum_{e=1}^{ne} \frac{\partial \phi_e(\mathbf{x}_{ck} - \mathbf{x}_e)}{\partial x_i} Y_{\alpha e} \quad (12.16)$$

where  $\mathbf{x}_e$  is the location of stencil point  $e$ ,  $Y_{\alpha e} = Y_\alpha(\mathbf{x}_e)$  and  $\phi_e(\mathbf{x} - \mathbf{x}_e)$ ,  $e = 1, \dots, ne$  is a suitable set of interpolation functions. For this discussion we assume that no boundary values of  $Y_\alpha$  are involved on the interpolation. For these faces the diffusive flux is:

$$(-\rho D_q \nabla Y_\alpha)_k = - \sum_{m=1}^{ne} \phi_m(\mathbf{x}_{ck} - \mathbf{x}_m) [\rho D_\alpha]_m \times \sum_{e=1}^{ne} \nabla \phi_e(\mathbf{x}_{ck} - \mathbf{x}_e) Y_{\alpha e}$$

$$= -[\rho D_\alpha]_k \sum_{e=1}^{ne} \nabla \phi_e(\mathbf{x}_{ck} - \mathbf{x}_e) Y_{\alpha e} \quad (12.17)$$

where

$$[\rho D_\alpha]_k = \left[ \sum_{m=1}^{ne} \phi_m(\mathbf{x}_{ck} - \mathbf{x}_m) [\rho D_\alpha]_m \right] \quad (12.18)$$

### 1D Linear Isoparametric Polynomial Interpolation:

Consider a stencil with cells  $jj$  and  $ii$  as in Fig. 12.1b. Their centroids are located at  $\mathbf{x}_{jj}$  and  $\mathbf{x}_{ii}$  respectively. Parameterizing point locations along the segment that unites  $\mathbf{x}_{jj}$  to  $\mathbf{x}_{ii}$  by  $-1 \leq \xi \leq 1$ , we have

$$\mathbf{x}(\xi) = \phi_1(\xi)\mathbf{x}_{jj} + \phi_2(\xi)\mathbf{x}_{ii} \quad (12.19)$$

$$Y_\alpha(\xi) = \phi_1(\xi)Y_{\alpha jj} + \phi_2(\xi)Y_{\alpha ii} \quad (12.20)$$

$$[\rho D_\alpha](\xi) = \phi_1(\xi)[\rho D_\alpha]_{jj} + \phi_2(\xi)[\rho D_\alpha]_{ii} \quad (12.21)$$

where

$$\phi_1(\xi) = \frac{1}{2}(1 - \xi); \phi_2(\xi) = \frac{1}{2}(1 + \xi) \quad (12.22)$$

We assume the value of the parameter where the cut-face lies known  $\xi_p$ , corresponding to the face and centroid segment intersection  $\mathbf{x}_p$ . From Eq. (12.19) with  $\mathbf{x} = x\hat{\mathbf{i}} + y\hat{\mathbf{j}}$ , we have

$$\frac{\partial \xi}{\partial x} = \frac{2}{(x_{ii} - x_{jj})}; \frac{\partial \xi}{\partial y} = \frac{2}{(y_{ii} - y_{jj})} \quad (12.23)$$

and the interpolation function gradient components are:

$$\frac{\partial \phi_1}{\partial x} = \frac{\partial \phi_1}{\partial \xi} \frac{\partial \xi}{\partial x} = -\frac{1}{(x_{ii} - x_{jj})} \quad (12.24)$$

$$\frac{\partial \phi_1}{\partial y} = \frac{\partial \phi_1}{\partial \xi} \frac{\partial \xi}{\partial y} = -\frac{1}{(y_{ii} - y_{jj})} \quad (12.25)$$

$$\frac{\partial \phi_2}{\partial x} = \frac{\partial \phi_2}{\partial \xi} \frac{\partial \xi}{\partial x} = \frac{1}{(x_{ii} - x_{jj})} \quad (12.26)$$

$$\frac{\partial \phi_2}{\partial y} = \frac{\partial \phi_2}{\partial \xi} \frac{\partial \xi}{\partial y} = \frac{1}{(y_{ii} - y_{jj})} \quad (12.27)$$

which are constant. Finally, being  $k = 3$ , for cut-cell  $jj$  the normal is  $\hat{\mathbf{n}}_{jj,k} = \hat{\mathbf{j}}$ , and expression (12.17) reduces to

$$(-\rho D_\alpha \nabla Y_\alpha)_k \cdot \hat{\mathbf{n}}_{jj,k} A_k = -[\rho D_\alpha]_p \frac{(Y_{\alpha ii} - Y_{\alpha jj})}{(y_{ii} - y_{jj})} A_k \quad (12.28)$$

Note that, an assumption of approximation of the cut-face centroid location by point  $p$  has been made. For cut-cell  $ii$  on the high side of face  $k = 3$  the expression is

$$(-\rho D_\alpha \nabla Y_\alpha)_k \cdot \hat{\mathbf{n}}_{ii,k} A_k = [\rho D_\alpha]_p \frac{(Y_{\alpha ii} - Y_{\alpha jj})}{(y_{ii} - y_{jj})} A_k \quad (12.29)$$

Similar expressions are found for a GASPHASE cut-face normal to the  $x$  direction.

### C. Faces 1, 2 are INBOUNDARY Cut-Faces:

Consider INBOUNDARY cut-face  $k = 1$  in Fig. 12.1b. If a Neumann boundary condition is specified in such a face, the term

$$(-\rho D_\alpha \nabla Y_\alpha)_k \cdot \hat{\mathbf{n}}_{ii,k} A_k = [\rho D_\alpha]_k \partial \gamma_{ii,k} A_k \quad (12.30)$$

where as before, the prescribed normal derivative of  $Y_\alpha$  on this face  $\partial Y_\alpha / \partial n|_k = \partial \gamma_{ii,k}$  is defined in the direction of  $\hat{\mathbf{n}}_1$ . This term adds to  $\mathbf{F}_{BC\alpha}$  in the location of cell  $ii$ . For SOLID\_BOUNDARIES, zero diffusive mass flux implies a homogeneous  $\partial Y_\alpha / \partial n|_k = 0$  for the mass fraction field of species  $\alpha$ . In case a Dirichlet boundary condition is specified in boundary face  $k = 1$  the normal mass fraction derivative is taken as:

$$\frac{\partial Y_\alpha}{\partial n}|_{k=1} = \frac{Y_{Ec1} - Y_{\alpha c1}}{h} \quad (12.31)$$

where the value of  $Y_{Ec1}$  is the cell value, and  $h$  is taken as half the cell volume divided by surface boundary area on the cell. Note that, as discussed later in Sec. 12.1.6, when small cells are linked to neighboring cells these quantities refer to the control volume of linked cells.

#### 12.1.5 Cut-Cell Discretization of Advective Term

We study the discretization of the advective term for the six faces of cell  $ii$  in Fig. 12.1b:

$$\int_{\Omega_{ii}} \nabla \cdot (\rho Y_\alpha \mathbf{u}) d\Omega = \int_{\partial \Omega_{ii}} (\rho Y_\alpha \mathbf{u}) \cdot \hat{\mathbf{n}}_{ii} d\partial\Omega = \sum_{k=1}^{nf_c} \left( \overline{[\rho Y_\alpha]} \mathbf{u} \right)_k \cdot \hat{\mathbf{n}}_{ii,k} A_k \quad (12.32)$$

where, as required for stability in the treatment of the advective term,  $\overline{[\rho Y_\alpha]}_k$  represents a flux limited interpolation to the face. See Sec. 3.2.1. For completeness, the use of interpolation functions in the interpolation process to a face, can also be extended to flux limited interpolation. Consider for a generic face  $k$  the advective flux for the conserved quantity  $\rho Y_\alpha$  off cell  $ii$ :

$$\left( \overline{[\rho Y_\alpha]} \mathbf{u} \right)_k \cdot \hat{\mathbf{n}}_{ii,k} = \overline{[\rho Y_\alpha]}_k (\mathbf{u}_k \cdot \hat{\mathbf{n}}_{ii,k}) \quad (12.33)$$

Then, a flux limited spatial interpolation of the form (12.15) for  $[\rho Y_\alpha]$  to face centroid  $ck$  is

$$\overline{[\rho Y_\alpha]}_k = \sum_{e=1}^{ne} \bar{\phi}_e(\mathbf{x}_{ck} - \mathbf{x}_e, \mathbf{u}_k) (\rho Y_\alpha)_e \quad (12.34)$$

where  $e = 1, \dots, ne$  refers to a suitable flux limited stencil of GASPHASE cell centroids. The  $\bar{\phi}_e$  are the flux limited interpolation functions related to said centroids.

### A. Faces $k = 4, 5$ are regular GASPHASE Faces Connecting a Cut-Cell with a Regular Cell:

The treatment of these is as described in the previous section, with the caveat that the advective flux computation for  $\rho Y_\alpha$  may now involve a four point stencil ( $e = 1, \dots, 4$ ) spatial flux matched interpolation to the face. Consider the situations depicted in Fig. 12.2 for a regular face  $RC$  connecting regular cell and cut-cell. In the left figure both extreme interpolation points of the stencil have associated species densities  $\rho Y_{\alpha 1}$  and  $\rho Y_{\alpha 4}$ , as at least one cut-cell or regular cells are defined at those locations. In the right figure the lower extreme position of the stencil (cell S) lies completely within the SOLID phase, and it takes the value of the inner cell  $\rho Y_{\alpha 2}$ , rendering the interpolation to  $RC$  of Godunov type. Note also the loss of accuracy on the flux limited interpolation described in Sec. 3.2.1 due to cut-cell centroids making the stencil not uniformly spaced.

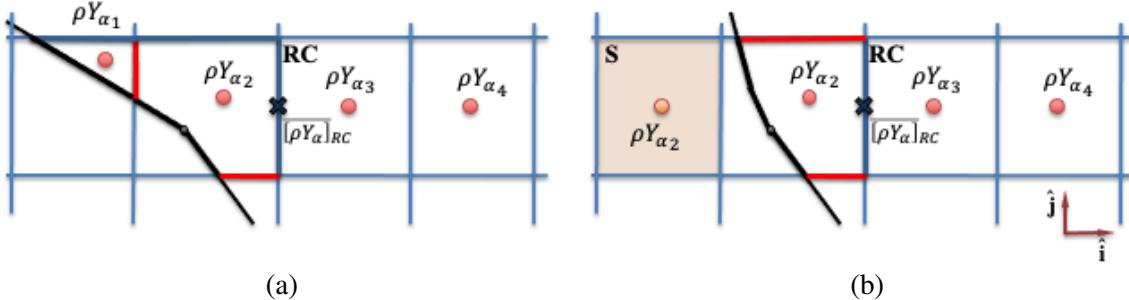


Figure 12.2: Flux limited interpolation: (a) Regular and cut-cell species densities are available for 4 point stencil flux limited interpolation to face  $RC$ . (b) Lower cell in stencil is within solid region (cell  $S$ ), therefore inner cell value  $\rho Y_2$  is used for this extreme point.

### B. Faces $k = 3, 6$ are GASPHASE Cut-Faces:

For GASPHASE cut-face  $k = 3$  in Fig. 12.1b, the advective term corresponding to cell  $jj$  is

$$(\rho Y_\alpha \mathbf{u})_k \cdot \hat{\mathbf{n}}_{jj,k} A_k = \overline{[\rho Y_\alpha]}_k (\mathbf{u}_k \cdot \hat{\mathbf{n}}_{jj,k}) A_k \quad (12.35)$$

where the average face normal velocity  $\mathbf{u}_k \cdot \hat{\mathbf{n}}_{jj,k}$  (i.e. approximated by the face centroid value) is assumed known. The quantity  $\overline{[\rho Y_\alpha]}_k$  is obtained by flux limited interpolation to the face in similar manner as described previously. The flux component for cut-cell  $ii$  is naturally opposite in sign to what is computed for cell  $jj$ .

### C. Faces 1, 2 are INBOUNDARY Cut-Faces:

The treatment of advective terms in boundary cut-faces is similar as described before. Consider a normal velocity  $\mathbf{u}_1 = v_n \hat{\mathbf{n}}_{ii,1}$  (into solid region) imposed at the boundary in INBOUNDARY cut-face  $k = 1$  in Fig. 12.1b. The advective component is in this case:

$$(\rho Y_\alpha \mathbf{u})_1 \cdot \hat{\mathbf{n}}_{ii,1} A_k = \overline{[\rho Y_\alpha]}_1 v_n A_k \quad (12.36)$$

and the value of  $\overline{[\rho Y_\alpha]}_1 = \rho Y_{\alpha ii}$  if  $v_n > 0$ , or  $\overline{[\rho Y_\alpha]}_1 = \rho \gamma_{ii,1}$  prescribed otherwise.

#### 12.1.6 Unsteady Evolution: Explicit Time Integration for Scalars

It is well known that cut-cell methods pose a significant time constraint when used with explicit time integration methods. In general, there will arise GASPHASE cut-cells whose small-size will severely penalize the time step. We recall that each cell on the gas phase, including cut-cells, needs to meet CFL and Von Neuman stability constraints. Several different ways have been proposed in the literature to deal with this problem, i.e., cell merging, mixing or linking methods. In general, these lead to ad hoc selection procedures for surrounding cells, having to deal with many special cases, and in some cases potential solution deterioration close to the boundary.

We use a simple and robust procedure to address this problem. Within the scheme for numeration of scalar cell unknowns, a test is performed on cut-cells. If the cut-cell volume is less than the threshold volume  $V_{thr} = \text{CCVOL\_LINK } V_{cart}$ , where  $V_{cart}$  is the local Cartesian cell volume and  $\text{CCVOL\_LINK} < 1$  is a threshold factor (default value 0.5), the unknown number this cell takes is the one of an adjacent cell which has a volume larger than  $V_{thr}$ . This mathematically defines a single control volume of the two linked cells. Cell volumes are added in building the mass matrix for the FV discretization, and fluxes and matrix terms

are added with their corresponding signs. Note that, flux quantities corresponding to the common face of these two cells effectively cancel on the single equation for the set. Alternatively, if after a number of cell numbering iterations (default 2) an unlinked small cell persists in the mesh, this small cell is blocked and its corresponding cartesian cell tagged as solid.

## 12.2 Momentum Time Marching and Immersed Boundaries

### 12.2.1 Momentum-Pressure Coupling

As a first approximation, consider the Newtonian flow problem defined by the following set of partial differential equations:

$$\frac{\partial \mathbf{u}(\mathbf{x}, t)}{\partial t} = -[\mathbf{F}(\mathbf{u}, \mathbf{x}, t) + \nabla H(\mathbf{x}, t)], \quad \mathbf{x} \in \Omega - \sum \Omega_i, \quad t \in \mathbb{R}_+ \quad (12.37)$$

$$\nabla \cdot \mathbf{u}(\mathbf{x}, t) = (\nabla \cdot \mathbf{u})^{th} \quad (12.38)$$

where equation (12.37) is the momentum equation,  $\mathbf{u}(\mathbf{x}, t)$  is the spatial velocity field,  $\mathbf{F}(\mathbf{u}, \mathbf{x}, t)$  is a vector containing convective, diffusive and possibly other force terms, and  $H(\mathbf{x}, t)$  is a potential scalar field (physically the head field in this case, commonly called pressure). For sake of argument here, to represent the low Mach number approximation employed in FDS, it suffices to consider a specified divergence field  $(\nabla \cdot \mathbf{u})^{th}(\mathbf{x}, t)$  (thermodynamic divergence). This divergence field in the thermally buoyant flow model used in FDS is a proxy for the energy equation. The domain  $\Omega - \sum \Omega_i$  represents the fluid region, and boundary conditions are prescribed for  $\mathbf{u}(\mathbf{x}, t)$  on  $\partial\Omega, \partial\Omega_1, \dots, \partial\Omega_{nbods}$ .

Classical fractional step methods for time integration of incompressible or low Mach number flow are based on two operations: First, momentum transport to obtain intermediate velocities, and second, projection of velocities into a target divergence field. Consider the Forward Euler update of the governing equations from  $t_n$  to  $t_{n+1} = t_n + \Delta t$  of the form: Given  $\mathbf{u}^n = \mathbf{u}(\mathbf{x}, t_n)$ ,  $\nabla \cdot \mathbf{u}^{n+1} = (\nabla \cdot \mathbf{u}^{n+1})^{th}$  known

$$\frac{\mathbf{u}^{n+1} - \mathbf{u}^n}{\Delta t} = -[\mathbf{F}^n + \nabla H^n] \quad (12.39)$$

$$\nabla \cdot \mathbf{u}^{n+1} = (\nabla \cdot \mathbf{u}^{n+1})^{th} \quad (12.40)$$

where as time has been discretized,  $\mathbf{u}^{n+1}$  represents a numerical approximation to the solution in Eqs. (12.37)-(12.38) at time  $t_{n+1}$ . As the potential field  $H(\mathbf{x}, t)$  does not have a time evolution equation, it is assumed responsible of enforcing the divergence condition and used on the projection step. Taking the divergence of equation (12.39) and considering the constraint Eq. (12.40), the two steps of the method are

1. Solve Poisson equation for  $H^n$ :

$$\nabla \cdot \nabla H^n = - \left[ \frac{(\nabla \cdot \mathbf{u}^{n+1})^{th} - \nabla \cdot \mathbf{u}^n}{\Delta t} \right] - \nabla \cdot \mathbf{F}^n \quad (12.41)$$

2. Obtain final velocity for step:

$$\mathbf{u}^{n+1} = \mathbf{u}^n - \Delta t [\mathbf{F}^n + \nabla H^n] \quad (12.42)$$

The term  $\hat{\mathbf{u}}^{n+1} = \mathbf{u}^n - \Delta t \mathbf{F}^n$  is known as intermediate velocity, and is a non matching divergence approximation to  $\mathbf{u}^{n+1}$  (i.e.  $\nabla \cdot \hat{\mathbf{u}}^{n+1} \neq (\nabla \cdot \mathbf{u}^{n+1})^{th}$ ).

Although in the original problem, Eqs. (12.37)-(12.38), no boundary condition is required for  $H(\mathbf{x}, t)$ , a consequence of the projection scheme is that boundary conditions are required on the Poisson equation of step Eq. (12.41). For explicit methods and stationary *solid* boundaries, the corresponding boundary condition is *homogeneous* Neumann for  $H^n, \partial H^n / \partial x_n = 0$  in  $\partial\Omega, \partial\Omega_1, \dots, \partial\Omega_{nbods}$  (i.e. [131]).

The next component of the scheme involves approximating velocity boundary conditions for immersed solid boundaries. In FDS, velocity field components are staggered on cartesian or cut faces. For complex geometry the projection scheme defined above is applied in the unstructured mesh composed by cut-cells and regular gas cells. Therefore, the Poisson equation is discretized in an unstructured grid and solved by matrix direct solvers, either by FDS mesh (ULMAT) or globally (UGLMAT). Currently the numerical scheme allows only one cut-cell/face cartesian per cell/face, due to the current need to define a single background pressure per cartesian cell. The blocking treatment of split cells is described in the Users guide. The projection step to obtain final velocities involves grid cell sizes in regular faces and centroid to centroid distance in cut-faces.

Cross velocities and tangent stresses and vorticity emanating from the immersed boundary are defined on cartesian faces being intersected by the geometry surface and surrounding cartesian edges. We use a wall modeled stress imposition method (STM) to this end described next.

### 12.2.2 Stress Method: Tangential Velocities Estimated Through Cartesian Stresses

In the discrete momentum equations, Eq. (12.39), the expression for  $\mathbf{F}^n$  is:

$$\mathbf{F}^n = - \left[ \mathbf{u} \times \boldsymbol{\omega} + \tilde{p} \nabla \left( \frac{1}{\rho} \right) + \frac{1}{\rho} [(\rho - \rho_0) \mathbf{g} + \nabla \cdot \boldsymbol{\tau}] \right]^n \quad (12.43)$$

In this section we are interested in the divergence of deviatoric stress  $\nabla \cdot \boldsymbol{\tau}$  term, its discretization and shear stress values at the boundary.

#### Wall Modeled Shear Stress for Grid Aligned Solid

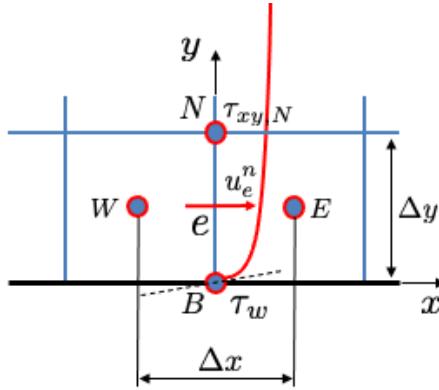


Figure 12.3: Wall stress  $\tau_w$  estimation using a wall function for solid boundary crossing through vertex  $B$  and aligned with the fluid grid.

Consider the grid aligned case of Fig. 12.3. A velocity component  $u_e^n$  is defined in a face adjacent to the solid boundary. The vertex point on the wall for this face is point  $B$ . A known shear stress  $\tau_w$  is provided at this boundary point  $B$ . In two dimensions, the  $x$  source component  $\nabla \cdot \boldsymbol{\tau}$  at point  $e$  can be computed as:

$$\left[ \frac{\partial \tau_{xx}}{\partial x} + \frac{\partial \tau_{xy}}{\partial y} \right]^n \simeq \left[ \frac{\tau_{xx,E} - \tau_{xx,W}}{\Delta x} + \frac{\tau_{xy,N} - \tau_w}{\Delta y} \right]^n \quad (12.44)$$

Therefore, in an explicit time integration scheme, a boundary value of shear stress  $\tau_w$  has to be estimated and used as source in the momentum equations. In FDS, this procedure is done using a wall model. A simple wall model representation in this case is given by a function  $u^+ = f(y^+)$ , and the near wall scaling relationships:

$$u_\tau = \sqrt{\frac{\tau_w}{\rho}}; \quad y^+ = \frac{y}{\delta_v}; \quad \delta_v = \frac{v}{u_\tau}; \quad u^+ = \frac{u}{u_\tau} \quad (12.45)$$

Here  $\tau_w$  is the stress at the wall point  $B$ ;  $\rho$  and  $v$  are the density and molecular kinematic viscosity. The quantity  $\delta_v$  is the near wall viscous length scale. The functional model used has been described in Sec. 4.4, and corresponds to a zero pressure gradient equilibrium turbulent boundary layer (*log-law*). As this function is non-linear,  $y^+$ ,  $\tau_w$ ,  $u_\tau$ , and  $\delta_v$  are computed to convergence through fixed point iteration.

Referring to Fig. 12.3, and considering the expression for the wall stress  $\tau_w = \rho v \left( \frac{\partial u}{\partial y} \right)_{y=0}$  and a known velocity value  $u_e^n$ , initialize:

- compute  $\tau_w^0 = \rho v \frac{2|u_e^n|}{\Delta y}$
- compute  $u_\tau^0 = \sqrt{\tau_w^0 / \rho}$
- compute  $\delta_v^0 = v / u_\tau^0$
- compute  $y^{+,0} = \frac{\Delta y}{2\delta_v^0}$

If a direct numerical simulation (DNS) is performed, this is all that is needed to compute the wall stress (its definition). For wall modeled large eddy simulation (LES), for  $k = 1, 2, \dots$  up to desired convergence do:

1. compute  $y^{+,k} = \frac{\Delta y}{2\delta_v^{k-1}}$
2. if  $y^{+,k} < 12$  : compute  $\tau_w^k = \rho (u_e^n / y^{+,k})^2$ ; else use  $\tau_w^k = \rho \left( \frac{u_e^n}{\ln(y^{+,k})/\kappa + Br} \right)^2$
3. compute  $u_\tau^k = \sqrt{\tau_w^k / \rho}$
4. compute  $\delta_v^k = v / u_\tau^k$

where  $\kappa \simeq 0.41$  and  $Br \simeq 5.2$  are the log law constants. Once the stress at the wall  $\tau_w$  is computed, it is used in the right hand side definition of the momentum equations to evolve the velocity flow variable  $u_e^n \rightarrow u_e^{n+1}$ . This velocity component off the wall is integrated in time in the same manner as the rest of the discrete velocities, having to obey CFL and viscous constraints for stability.

### Non Grid Aligned Surfaces: Gradient Estimation in Cartesian Directions

When the solid surface is not aligned with the fluid mesh, velocity gradients and cartesian shear stress components are computed in the body surface and then are interpolated or extrapolated to surrounding cartesian grid edges. In general, the approach taken is similar to what is presented in references [132, 133], but in our case, sampling points for the different cartesian gradient components are defined in the cartesian instead of wall normal directions. In Fig. 12.4a, we define inside-boundary (IB) edges as edges contained completely in the solid (or solid surface) but adjacent to the boundary. Adjacency is defined as having one of the edge connected cartesian faces being intersected by the solid boundary. We also define regular-cut face (RC) edges as edges completely defined in the gas phase, but adjacent to the boundary. Edges defined as IB or RC will receive vorticity and shear stress components extrapolated or interpolated from values estimated

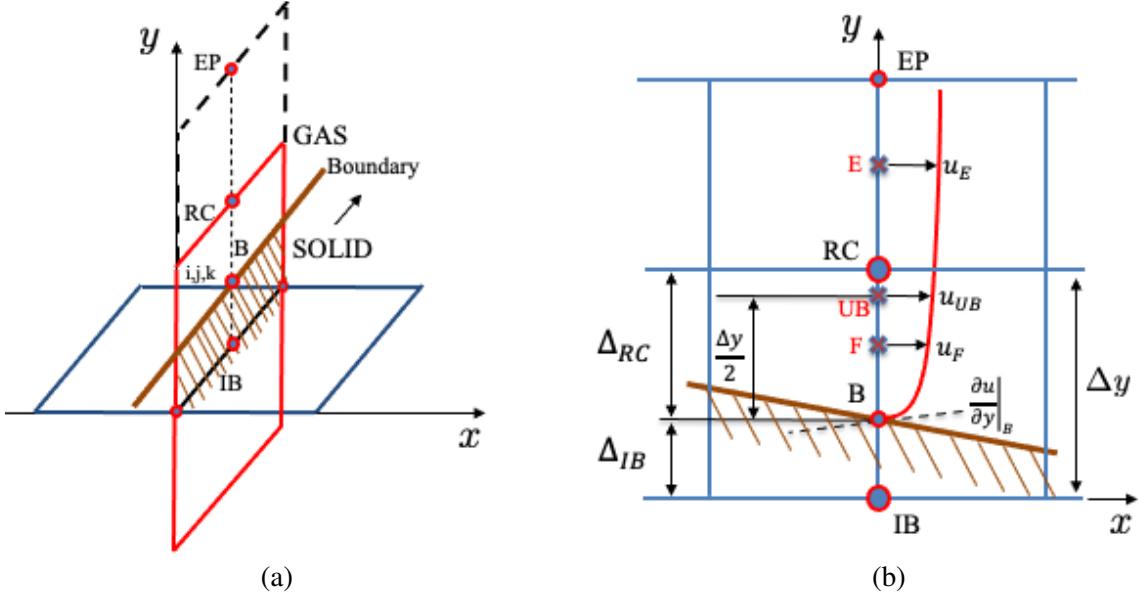


Figure 12.4: Velocity gradients and stress components estimation. (a) For cut-face  $(i,j,k)$  velocity gradient  $\partial u / \partial y$  and shear stress component  $\mu \partial u / \partial y$  are estimated using a wall function for solid boundary crossing through vertex  $B$ , and transferred to cartesian edges  $IB$ ,  $RC$ . (b) Details of wall function used in obtaining cartesian velocity derivative, shear stress component.

in the boundary through a wall model and gas phase values. The procedure might require an external gas phase edge (EP), also depicted in the figure.

For each cartesian face containing  $IB$  and  $RC$  edges, shear stress components are modified to account for the presence of the boundary. We assume the average location  $B$  of the boundary is known in each direction for each cartesian face containing cut-faces. In the following, all variables are assumed at time level  $n$ . In the case of  $x$ -face  $(i,j,k)$  in the figure,  $\tau_{xy,IB}$  and  $\tau_{xy,RC}$  are redefined as described in the following:

1. For each  $RC$  edge compute the gas finite difference estimation of the cross velocity derivatives  $\partial u_i / \partial x_j$  and  $\partial u_j / \partial x_i$ . In Fig. 12.4b,  $\frac{\partial u}{\partial y} \simeq 2 \frac{u_E - u_F}{\Delta_y + \Delta_{RC}}$ , where  $u_E = u_{i,j+1,k}$  and  $u_F$  is the cut-face velocity, and  $\frac{\partial v}{\partial x} \simeq \frac{v_{i+1,j,k} - v_{i,j,k}}{\Delta_x}$ . Note that the distance to the wall  $\Delta_{RC}$  is being used to compute the finite difference in the cut-face and  $y$  direction. Also interpolating the dynamic viscosity to the  $RC$  edge, estimate  $\mu_{RC} \frac{\partial u}{\partial y}$  and  $\mu_{RC} \frac{\partial v}{\partial x}$ .
2. Focusing in the face containing  $IB$  and  $RC$  edges, define point  $UB$  at a distance  $\Delta y/2$  from the boundary  $B$ . Obtain velocity component  $u_{UB}$  at location  $UB$  by linear interpolation of  $u_E$  and  $u_F$  in the  $y$  direction.
3. Using the wall function compute  $\partial u / \partial y$  and  $\mu \partial u / \partial y$  at boundary point  $B$ . Note that the input into the wall function routine is the velocity component  $u_{UB}$  a distance  $\Delta_y/2$  along the cartesian direction  $y$ . We are using the wall model in a direction other than the wall normal direction.
4. The external edge EP is defined as: Take EP to be the  $RC$  edge if  $\Delta_{IB} \leq \Delta y/2$ , use edge out in the cartesian direction  $(i,j+1,k)$  otherwise. Compute  $\partial u / \partial y$  and  $\mu \partial u / \partial y$  at EP.
5. Linearly extrapolate  $\partial u / \partial y$  and  $\mu \partial u / \partial y$  to  $IB$  edge from boundary  $B$  and EP values.
6. If EP is the outer edge to  $RC$ , linearly interpolate  $\partial u / \partial y$  and  $\mu \partial u / \partial y$  to  $RC$  from EP and  $B$  values. To allow for  $u_F$  feedback on the computation of this quantity we average the interpolated  $\partial u / \partial y$  and  $\mu \partial u / \partial y$  with the value computed in point 1.

Finally, all RC and IB edges are assigned a vorticity and shear stress using the computed gradients  $\partial u_i / \partial x_j$  and components  $\mu \partial u_i / \partial x_j$ . The computed vorticity and stresses are used in the construction of the force vector  $\mathbf{F}^n$  of equation (12.43) and used in the projection equations (12.41)-(12.42).

In similar manner as in scalar transport, small cut-faces and their associated volumes will constraint the momentum explicit time integration steps for stability. A similar linking procedure as described for scalars is employed to link small cut-faces to surrounding faces and to define a single average value of  $\bar{\mathbf{F}}^n$  for the linked set. Pressure gradient values on these faces are not linked, as required for correct cell by cell enforcement of the discrete divergence constraint and mass conservation.

## 12.3 Energy Equation, Thermodynamic Divergence

### 12.3.1 Divergence Constraint

Starting from the sensible enthalpy evolution equation, the divergence of the velocity field imposed in eq. (12.38) can be factored as:

$$\begin{aligned} (\nabla \cdot \mathbf{u})^{th} &= \left[ \frac{1}{\rho c_p T} - \frac{1}{\bar{p}} \right] \frac{\partial \bar{p}}{\partial t} + \frac{w \rho_0 g_z}{\rho c_p T} \\ &+ \frac{1}{\rho c_p T} [\dot{q}''' - \nabla \cdot \dot{\mathbf{q}}'' - \mathbf{u} \cdot \nabla (\rho h_s)] \\ &+ \frac{1}{\rho} \sum_{\alpha} \left( \frac{\bar{W}}{W_{\alpha}} - \frac{h_{s,\alpha}}{c_p T} \right) [\dot{m}_{\alpha}''' - \nabla \cdot \mathbf{J}_{\alpha} - \mathbf{u} \cdot \nabla (\rho Y_{\alpha})] \end{aligned} \quad (12.46)$$

we call this divergence expression the thermodynamic divergence  $(\nabla \cdot \mathbf{u})^{th}$ . The projection scheme for velocities enforces this final divergence on the discrete velocity field in each cell of the spatial discretization. Next, we look at the finite volume discretization of the non-conservative advection terms involving the velocity  $\mathbf{u}$ , in the right hand side of Eq. (12.46).

We will use the finite volume version of the previous equation to obtain the volume integrated target thermodynamic divergence on each cut cell volume, and to recompute the thermodynamic divergence on regular gas phase cells adjacent to cut-cells.

### 12.3.2 Finite-Volume Discretization of Scalar Advection Terms

Consider the the equality

$$\nabla \cdot \phi \mathbf{u} = \mathbf{u} \cdot \nabla (\phi) + \phi \nabla \cdot \mathbf{u} \quad (12.47)$$

Integrating over the volume of cell  $ii$

$$\int_{\Omega_{ii}} \mathbf{u} \cdot \nabla (\phi) d\Omega = \int_{\Omega_{ii}} \nabla \cdot (\phi \mathbf{u}) d\Omega - \int_{\Omega_{ii}} \phi \nabla \cdot \mathbf{u} d\Omega \quad (12.48)$$

Assuming a flux limited interpolation of the scalar  $\phi$  at the cell boundaries, the discrete counterpart is

$$\overline{\mathbf{u} \cdot \nabla (\phi)} V_{ii} = \sum_{k=1}^{nf_c} (\overline{\phi} \mathbf{u})_{ii,k} \cdot \hat{\mathbf{n}}_{ii,k} A_k - [\phi]_{ii} \sum_{k=1}^{nf_c} \mathbf{u}_{ii,k} \cdot \hat{\mathbf{n}}_{ii,k} A_k \quad (12.49)$$

Here the overline states that the cell-centered scalar quantity  $\phi \in \{\rho h_s, \rho Y_{\alpha}\}$  has been interpolated to the cell faces using a flux limited interpolation scheme. The overline in  $\mathbf{u} \cdot \nabla (\phi)$ , states that this term is computed consistent with the flux limited interpolation adopted. Also,  $V_{ii}$  is the volume of cell  $ii$ , and  $A_k$  is the area of a face  $k = 1, \dots, nf_c$  of the given cell. This last equation is the finite volume counterpart of Eq. (B.12).

### 12.3.3 Discrete Thermodynamic Divergence Expression

For cut-cell  $ii$ , the corresponding discrete volume integrated expression is:

$$\begin{aligned} (\nabla \cdot \mathbf{u})_{ii}^{th} V_{ii} &= \left[ \frac{1}{(\rho c_p T)_{ii}} - \frac{1}{\bar{p}_{ii}} \right] \frac{\partial \bar{p}_{ii}}{\partial t} V_{ii} + \frac{w_{ii} \rho_0 g_z}{(\rho c_p T)_{ii}} V_{ii} \\ &+ \frac{1}{(\rho c_p T)_{ii}} \left[ \dot{q}''' V_{ii} - \nabla \cdot \dot{\mathbf{q}}''_r V_{ii} - \sum_{k=1}^{nf_c} \dot{\mathbf{q}}''_{ii,k} \cdot \hat{\mathbf{n}}_{ii,k} A_k - \overline{\mathbf{u} \cdot \nabla(\rho h_s)} V_{ii} \right] \\ &+ \frac{1}{\rho_{ii}} \sum_{\alpha} \left( \frac{\overline{W}}{W_{\alpha}} - \frac{h_{s,\alpha}}{c_p T} \right)_{ii} \left[ \dot{m}_{\alpha}''' V_{ii} - \sum_{k=1}^{nf_c} \mathbf{J}_{\alpha,ii,k} \cdot \hat{\mathbf{n}}_{ii,k} A_k - \overline{\mathbf{u} \cdot \nabla(\rho Y_{\alpha})} V_{ii} \right] \end{aligned} \quad (12.50)$$

where the overline terms refer to flux limited interpolation of corresponding scalars, terms defined with subscript  $ii$  refer to cell defined quantities, and the scalar diffusive flux  $\mathbf{J}_{\alpha} = -\rho D_{\alpha} \nabla Y_{\alpha}$ . The divergence of radiative heat flux  $\nabla \cdot \dot{\mathbf{q}}''_r$  is provided by the solution of radiative transport equation, Sec. 6.2. Also, the vertical velocity  $w_{ii}$  has been interpolated to the cut-cell centroid. The flux vector  $\dot{\mathbf{q}}''_{ii,k}$  refers to diffusive and conductive heat fluxes described next.

### 12.3.4 Computing Diffusive Heat Flux in Finite-Volume Form

Given the species diffusion model adopted, the diffusive heat flux vector field is  $\dot{\mathbf{q}}''_{d,\alpha} = -h_s \rho D_{\alpha} \nabla(Y_{\alpha})$ . The integral of its divergence over a cell  $ii$  is:

$$\int_{\Omega_{ii}} \nabla \cdot (-h_{s,\alpha} \rho D_{\alpha} \nabla(Y_{\alpha})) d\Omega = \sum_{k=1}^{nf_c} (-h_{s,\alpha} \rho D_{\alpha} \nabla(Y_{\alpha}))_{ii,k} \cdot \hat{\mathbf{n}}_{ii,k} A_k \quad (12.51)$$

where  $(-h_s \rho D_{\alpha} \nabla(Y_{\alpha}))_{ii,k}$  is the mean heat flux defined on boundary surface  $k$  of cell  $ii$ .

### 12.3.5 Computing Heat Conduction Flux in Finite-Volume Form

The heat conduction flux vector of the mixture is defined as  $\dot{\mathbf{q}}''_{kT} = -k \nabla(T)$ , where  $k$  is the mixture conductivity. The integral of its divergence over a cell  $ii$  is computed similarly as in the previous section:

$$\int_{\Omega_{ii}} \nabla \cdot (-k \nabla(T)) d\Omega = \sum_{k=1}^{nf_c} (-k \nabla(T))_{ii,k} \cdot \hat{\mathbf{n}}_{ii,k} A_k \quad (12.52)$$

where  $(-k \nabla(T))_{ii,k}$  is the mean heat flux defined on boundary surface  $k$  of cell  $ii$ .



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## Appendix A

### Nomenclature

$A_{p,c}$	droplet/particle cross sectional area
$A_{p,s}$	droplet/particle surface area
$A_{\alpha\beta}$	pre-exponential factor for solid phase Arrhenius reaction
$B$	emission source term; particle mobility
$C$	sprinkler C-factor; coefficient of natural convection
$C_d$	drag coefficient
$C_m$	momentum accommodation coefficient
$C_n$	Cunningham slip correction factor
$C_s$	Smagorinsky constant (LES); thermal slip coefficient
$C_t$	thermal accommodation coefficient
$c$	solid material specific heat; speed of light in vacuum
$c_p$	constant-pressure specific heat
$D$	droplet/particle diameter
$D_\alpha$	diffusion coefficient
$D_{v,0.5}$	median volumetric droplet diameter
$E$	activation energy
$F_u$	Fuchs factor
$\mathbf{f}_b$	external force vector (excluding gravity)
$g$	acceleration of gravity
$\mathbf{g}$	gravity vector, normally $(0, 0, -g)$
$H$	total pressure divided by the density (Bernoulli integral)
$H_{r,\alpha\beta}$	heat of reaction for a solid phase reaction
$h$	heat transfer coefficient; mass transfer coefficient; enthalpy; Planck constant
$h_{s,\alpha}$	sensible enthalpy of species $\alpha$
$I$	radiation intensity per unit of solid angle
$I_b$	radiation blackbody intensity per unit of solid angle
$I_{b,\lambda}$	spectral radiation blackbody intensity as function of wavelength per unit of solid angle
$I_{b,\omega}$	spectral radiation blackbody intensity as function of wavenumber per unit of solid angle
$k$	thermal conductivity; suppression decay factor
$k_B$	Boltzmann constant
$Kn$	Knudsen number
$K_{th}$	thermophoretic velocity coefficient
$L$	characteristic length; surface thickness
$\dot{m}_{b,\alpha}'''$	mass production rate per unit volume of species $\alpha$ by evaporating droplets/particles

$\dot{m}_{\alpha}''$	mass production rate per unit volume of species $\alpha$ by chemical reactions
$\dot{m}_w''$	water mass flux
$m_w''$	water mass per unit area
$n_s$	partial reaction order for solid
$n_{O_2}$	partial reaction order for oxygen
Nu	Nusselt number
Pr	Prandtl number
$p$	pressure
$\bar{p}_0$	atmospheric pressure profile
$\bar{p}_m$	background pressure of $m$ th pressure zone
$\tilde{p}$	pressure perturbation
$\dot{\mathbf{q}}''$	heat flux vector
$\dot{q}'''$	heat release rate per unit volume
$\dot{q}_r''$	radiative heat flux
$\dot{q}_c''$	convective heat flux
$\dot{Q}$	total heat release rate
$\dot{Q}^*$	fire Froude number
$R$	universal gas constant
Re	Reynolds number
$r_p$	particle/droplet radius
$r_{\alpha\beta}$	solid phase reaction rate
RTI	Response Time Index of a sprinkler
$\mathbf{s}$	unit vector in direction of radiation intensity
Sc	Schmidt number
Sh	Sherwood number
$S_{\alpha}$	solid component production rate
$T$	temperature
$t$	time
$U$	integrated radiant intensity; optical pathlength
$\mathbf{u} = (u, v, w)$	velocity vector
$W_{\alpha}$	molecular weight of gas species $\alpha$
$\bar{W}$	molecular weight of the gas mixture
We	Weber number
$\mathbf{x} = (x, y, z)$	position vector
$X_{\alpha}$	volume fraction of species $\alpha$
$Y_{\alpha}$	mass fraction of species $\alpha$
$\bar{Y}_{\alpha}$	mean mass fraction of species $\alpha$
$\hat{Y}_{\alpha}$	mass fraction of species $\alpha$ in mixed zone of a computation cell
$Y_{O_2}^{\infty}$	mass fraction of oxygen in the ambient
$Y_F$	mass fraction of fuel
$y_s$	soot yield
$Z_{\alpha}$	species mixture $\alpha$

## Greek Letters

$\alpha$	ratio of gas conductivity to the particle conductivity; integrated band intensity
$\gamma$	ratio of specific heats; Rosin-Rammler exponent; spectral fine structure parameter of narrow band
$\Delta$	LES filter width

$\Delta h$	heat of combustion
$\Delta h_{O_2}$	energy released per unit mass oxygen consumed
$\Delta h_{f,\alpha}^0$	heat of formation of species $\alpha$
$\Delta h_g$	heat of gasification
$\delta$	film thickness; scaling factor of thickness and density
$\varepsilon$	dissipation rate; emissivity; agglomeration factor
$\eta$	agglomeration apportioning factor
$\kappa$	absorption coefficient; von Karman constant
$\lambda$	mean free path of gas molecules; wavelength of thermal radiation
$\mu$	dynamic viscosity
$\nu$	frequency of the thermal radiation
$v_\alpha$	stoichiometric coefficient, species $\alpha$
$v_s$	yield of solid residue in solid phase reaction
$v_{g,\gamma}$	yield of gaseous species $\gamma$ in solid phase reaction
$\rho$	density
$\bar{\tau}_\omega$	mean spectral transmissivity of a narrow band centered in $\omega$
$\tau_{ij}$	viscous stress tensor
$\Phi$	agglomeration kernel
$\phi$	porosity
$\chi, C_s$	shape factor
$\chi_r$	radiative loss fraction
$\sigma$	Stefan-Boltzmann constant; constant in droplet size distribution; surface tension
$\sigma_p$	particle scattering coefficient
$\sigma_s$	scattering coefficient
$\tau^+$	dimensionless stopping distance
$\omega$	wavenumber of thermal radiation
$\boldsymbol{\omega} = (\omega_x, \omega_y, \omega_z)$	vorticity vector
$\Omega$	solid angle

## Subscripts

0	initial value
a	air
b	bulk phase property; boiling
B	Brownian
c	convective
d	drag
e	effective properties
g	gas
$ijk$	gas phase cell indices
n	band properties
p	particle/droplet
PK	collision efficiency
p	pressure
r	radiative
s	solid; sensible; soot
S	sticking factor

w	wall
$\alpha$	gas species index
$\beta$	index of reaction
$\lambda$	wavelength
$\omega$	wavenumber

## Appendix B

# A Velocity Divergence Constraint for Large-Eddy Simulation of Low-Mach Flows

The equations governing the evolution of a low-Mach number, variable density fluid—first introduced by Rehm and Baum in 1978 [9]—are continuity, species concentration (mass fraction), momentum, energy (sensible enthalpy), and the ideal gas equation of state:

$$\frac{\partial \rho}{\partial t} + \nabla \cdot (\rho \mathbf{u}) = \dot{m}_b''' \quad (B.1)$$

$$\frac{\partial \rho Y_\alpha}{\partial t} + \nabla \cdot (\rho Y_\alpha \mathbf{u}) = \nabla \cdot (\rho D_\alpha \nabla Y_\alpha) + \dot{m}_\alpha'''' + \dot{m}_{b,\alpha}''' \quad (B.2)$$

$$\frac{\partial \rho \mathbf{u}}{\partial t} + \nabla \cdot (\rho \mathbf{u} \mathbf{u}) = -\nabla \tilde{p} - \nabla \cdot \boldsymbol{\tau} + (\rho - \rho_0) \mathbf{g} \quad (B.3)$$

$$\frac{\partial \rho h_s}{\partial t} + \nabla \cdot (\rho h_s \mathbf{u}) = \frac{D\bar{p}}{Dt} + \dot{q}'''' - \nabla \cdot \dot{\mathbf{q}}'' \quad (B.4)$$

$$\rho = \frac{\bar{p}\bar{W}}{RT} \quad (B.5)$$

In this appendix, starting from the conservative form of the sensible enthalpy transport equation, we derive a numerically consistent velocity divergence constraint for use in large-eddy simulation (LES) of low-Mach flows. The result accounts for numerical transport of mass and energy, which is difficult to eliminate in relatively coarse, engineering LES calculations when total variation diminishing (TVD) scalar transport schemes are employed. Without the correction terms derived here, unresolved (numerical) mixing of gas species with different heat capacities and molecular weights may lead to erroneous mixture temperatures and ultimately to an imbalance in the energy budget.

### B.1 The Divergence Constraint

As mentioned, the present work stems from attempts to understand and correct an energy budget imbalance that became evident after implementing both temperature-dependent specific heats and TVD scalar transport. One of the revelations of this work has been that the choice of starting point for deriving the divergence constraint naturally leads to two different forms of the divergence expression. While these forms are mathematically equivalent, they lead to two completely different—and yet completely plausible—numerical formulations.

### B.1.1 From Continuity

Starting from the continuity equation, we can factor out the velocity divergence leaving the material derivative of the density:

$$\nabla \cdot \mathbf{u} = -\frac{1}{\rho} \frac{D\rho}{Dt} + \frac{1}{\rho} \dot{m}_b''' \quad (\text{B.6})$$

Using the ideal gas law and differentiating the equation of state leads to

$$\begin{aligned} \nabla \cdot \mathbf{u} &= \left( \frac{1}{\rho c_p T} - \frac{1}{\bar{p}} \right) \frac{D\bar{p}}{Dt} \\ &\quad + \frac{1}{\rho c_p T} [\dot{q}''' - \nabla \cdot \dot{\mathbf{q}}''] \\ &\quad + \frac{1}{\rho} \sum_{\alpha} \left( \frac{\bar{W}}{W_{\alpha}} - \frac{h_{s,\alpha}}{c_p T} \right) [\nabla \cdot (\rho D_{\alpha} \nabla Y_{\alpha}) + \dot{m}_{\alpha}'''] \\ &\quad + \frac{1}{\rho} \sum_{\alpha} \left( \frac{\bar{W}}{W_{\alpha}} - \frac{\int_{T_b}^T c_{p,\alpha}(T') dT'}{c_p T} \right) \dot{m}_{b,\alpha}''' \end{aligned} \quad (\text{B.7})$$

### B.1.2 From Sensible Enthalpy

Alternatively, we may factor the velocity divergence from the sensible enthalpy transport equation:

$$\nabla \cdot \mathbf{u} = \frac{1}{\rho h_s} \left[ \frac{D}{Dt} (\bar{p} - \rho h_s) + \dot{q}''' - \nabla \cdot \dot{\mathbf{q}}'' \right] \quad (\text{B.8})$$

From this starting point, (arguably) the natural result for the divergence expression is

$$\begin{aligned} \nabla \cdot \mathbf{u} &= \frac{1}{\rho c_p T} \frac{D\bar{p}}{Dt} - \frac{1}{\bar{p}} \frac{\partial \bar{p}}{\partial t} \\ &\quad + \frac{1}{\rho c_p T} [\dot{q}''' - \nabla \cdot \dot{\mathbf{q}}'' - \mathbf{u} \cdot \nabla (\rho h_s)] \\ &\quad + \frac{1}{\rho} \sum_{\alpha} \left( \frac{\bar{W}}{W_{\alpha}} - \frac{h_{s,\alpha}}{c_p T} \right) [\nabla \cdot (\rho D_{\alpha} \nabla Y_{\alpha}) - \mathbf{u} \cdot \nabla (\rho Y_{\alpha}) + \dot{m}_{\alpha}'''] \\ &\quad + \frac{1}{\rho} \sum_{\alpha} \left( \frac{\bar{W}}{W_{\alpha}} - \frac{\int_{T_b}^T c_{p,\alpha}(T') dT'}{c_p T} \right) \dot{m}_{b,\alpha}''' \end{aligned} \quad (\text{B.9})$$

### B.1.3 Comparison

Notice the subtle differences between the first, second, and third lines of (B.7) and (B.9). The first lines differ by  $(\mathbf{u} \cdot \nabla \bar{p})/\bar{p}$ . In (B.9), the second and third lines each contain an extra term accounting for advection of enthalpy and mass, respectively,  $\mathbf{u} \cdot \nabla (\rho h_s)$  and  $\mathbf{u} \cdot \nabla (\rho Y_{\alpha})$ . Using (B.1)-(B.5), it can be shown that (B.7) and (B.9) are mathematically equivalent (see Section B.4).

## B.2 The Discrete Divergence

The *conservative form* of the sensible enthalpy transport equation—which derives its name from the flux divergence form of the mean transport term on the left hand side—is

$$\frac{\partial(\rho h_s)}{\partial t} + \underbrace{\nabla \cdot (\rho h_s \mathbf{u})}_{\text{mean transport}} = \frac{D\bar{p}}{Dt} + \dot{q}''' - \nabla \cdot \dot{\mathbf{q}}''. \quad (\text{B.10})$$

This form is called conservative because, by Gauss's theorem, the integral of the discrete flux divergence over the domain is equivalent to the surface integral of the flux over the boundary of the domain. For a periodic domain the integral is zero—*flow in* must equal *flow out*. The key to guaranteeing discrete conservation of sensible enthalpy is to first discretize the mean transport term. Below an overline will denote a slope-limiting interpolation operator. As discussed in Section B.2.1, this operator is specially designed to be consistent with flux-limited, total variation diminishing (TVD) transport for the conservative form of the mean transport term.

Expanding the mean transport term and rearranging (B.10) in terms of the discrete divergence yields

$$\nabla \cdot \mathbf{u} = \frac{1}{\rho h_s} \left[ - \left( \frac{\partial(\rho h_s)}{\partial t} + \overline{\mathbf{u} \cdot \nabla(\rho h_s)} \right) + \frac{D\bar{p}}{Dt} + \dot{q}''' - \nabla \cdot \dot{\mathbf{q}}'' \right]. \quad (\text{B.11})$$

The numerical details of  $\overline{\mathbf{u} \cdot \nabla(\rho h_s)}$  are the key to assuring discrete conservation (see Section B.2.1). Mathematically, (B.11) is equivalent to (B.8). Numerically, however, (B.11) accounts for the critical details of the TVD transport scheme.

Most of the complexity in the divergence expression is buried in the time derivative term,  $\partial(\rho h_s)/\partial t$ . Using (B.1)-(B.5), it can be shown that (B.11) expands to yield (B.9) (see Section B.3).

### B.2.1 Factoring the Discrete Flux Divergence

Below we show the numerical decomposition of the enthalpy flux divergence for cell  $i$  in 1D. The operator  $\delta(\ )/\delta x$  denotes a central difference. Density  $\rho$  and sensible enthalpy  $h_s$  are stored at cell centers indexed by  $i$ ,  $i+1$ , etc. Velocity  $u$  is stored at the cell face and indexed by  $i+\frac{1}{2}$ , etc. Here an overline applied to a face value ( $i \pm \frac{1}{2}$  suffix) denotes a flux limiter, which is basically a special interpolation of the scalar field to the cell face. The purpose of the flux limiter is to prevent spurious oscillations in the scalar solution. Such oscillations must be avoided because they may lead to boundedness violations and instability.

In decomposing the flux divergence, our goal is to break the term into two parts as follows:

$$\begin{aligned} \left[ \frac{\delta(\rho h_s u)}{\delta x} \right]_i &= \frac{\overline{(\rho h_s)}_{i+\frac{1}{2}} u_{i+\frac{1}{2}} - \overline{(\rho h_s)}_{i-\frac{1}{2}} u_{i-\frac{1}{2}}}{\delta x} \\ &= (\rho h_s)_i \underbrace{\frac{u_{i+\frac{1}{2}} - u_{i-\frac{1}{2}}}{\delta x}}_{\nabla \cdot \mathbf{u}} + \underbrace{\frac{\Delta_{i+\frac{1}{2}} u_{i+\frac{1}{2}} + \Delta_{i-\frac{1}{2}} u_{i-\frac{1}{2}}}{\delta x}}_{\overline{\mathbf{u} \cdot \nabla(\rho h_s)}} \end{aligned} \quad (\text{B.12})$$

Here  $\Delta_{i+\frac{1}{2}}$  represents a limited slope of the scalar data ( $\rho h_s$  in this case) at the face  $i+\frac{1}{2}$ . The slope limiters for cell  $i$  are defined such that

$$(\rho h_s)_i + \Delta_{i+\frac{1}{2}} = \overline{(\rho h_s)}_{i+\frac{1}{2}} \quad (\text{B.13})$$

$$(\rho h_s)_i - \Delta_{i-\frac{1}{2}} = \overline{(\rho h_s)}_{i-\frac{1}{2}} \quad (\text{B.14})$$

Note that while scalar face values are unique to the face  $\left[ \overline{(\rho h_s)}_{i+\frac{1}{2}} = \overline{(\rho h_s)}_{i+1-\frac{1}{2}} \right]$ , the limited slopes are not ( $\Delta_{i+\frac{1}{2}} \neq \Delta_{i+1-\frac{1}{2}}$ ).

### B.2.2 Example: Pure Upwinding

Suppose all  $u > 0$  in 1D, a wind from left to right. For Godunov's scheme (first-order upwinding) the limited slopes would be computed as follows:

$$\begin{aligned}\Delta_{i+\frac{1}{2}} &= \overline{(\rho h_s)}_{i+\frac{1}{2}} - (\rho h_s)_i \\ &= (\rho h_s)_i - (\rho h_s)_i \\ &= 0\end{aligned}\tag{B.15}$$

$$\begin{aligned}\Delta_{i-\frac{1}{2}} &= (\rho h_s)_i - \overline{(\rho h_s)}_{i-\frac{1}{2}} \\ &= (\rho h_s)_i - (\rho h_s)_{i-1}\end{aligned}\tag{B.16}$$

The cell-average advection term therefore becomes

$$\overline{\mathbf{u} \cdot \nabla(\rho h_s)} = u_{i-\frac{1}{2}} \left[ \frac{(\rho h_s)_i - (\rho h_s)_{i-1}}{\delta x} \right]\tag{B.17}$$

### B.2.3 Example: Central Differencing

For central differencing the limited slopes would be computed as follows:

$$\begin{aligned}\Delta_{i+\frac{1}{2}} &= \overline{(\rho h_s)}_{i+\frac{1}{2}} - (\rho h_s)_i \\ &= \frac{1}{2} [(\rho h_s)_i + (\rho h_s)_{i+1}] - (\rho h_s)_i \\ &= \frac{1}{2} [(\rho h_s)_{i+1} - (\rho h_s)_i]\end{aligned}\tag{B.18}$$

$$\begin{aligned}\Delta_{i-\frac{1}{2}} &= (\rho h_s)_i - \overline{(\rho h_s)}_{i-\frac{1}{2}} \\ &= (\rho h_s)_i - \frac{1}{2} [(\rho h_s)_{i-1} + (\rho h_s)_i] \\ &= \frac{1}{2} [(\rho h_s)_i - (\rho h_s)_{i-1}]\end{aligned}\tag{B.19}$$

The cell-average advection term therefore becomes

$$\overline{\mathbf{u} \cdot \nabla(\rho h_s)} = \frac{1}{2} u_{i+\frac{1}{2}} \left[ \frac{(\rho h_s)_{i+1} - (\rho h_s)_i}{\delta x} \right] + \frac{1}{2} u_{i-\frac{1}{2}} \left[ \frac{(\rho h_s)_i - (\rho h_s)_{i-1}}{\delta x} \right]\tag{B.20}$$

### B.2.4 General Implementation: Using Flux Limiters

The examples above are for illustration purposes only. In general, we first compute the flux-limited face values and obtain the limited slopes from (B.13) and (B.14). The cell-average advection term is then computed from the second underbrace in (B.12).

## B.3 Decomposing the Time Derivative

Using the ideal gas law, the time derivative of the enthalpy can be decomposed as follows:

$$\frac{\partial(\rho h_s)}{\partial t} = \rho \frac{\partial h_s}{\partial t} + h_s \frac{\partial \rho}{\partial t}$$

$$\begin{aligned}
&= \rho \sum_{\alpha} \left( Y_{\alpha} c_{p,\alpha} \frac{\partial T}{\partial t} + h_{s,\alpha} \frac{\partial Y_{\alpha}}{\partial t} \right) + h_s \frac{\partial \rho}{\partial t} \\
&= \rho c_p \frac{\partial T}{\partial t} + \rho \sum_{\alpha} h_{s,\alpha} \frac{\partial Y_{\alpha}}{\partial t} + h_s \frac{\partial \rho}{\partial t} \\
&= \rho c_p T \left[ \frac{1}{\bar{p}} \frac{\partial \bar{p}}{\partial t} + \frac{1}{\bar{W}} \frac{\partial \bar{W}}{\partial t} - \frac{1}{\rho} \frac{\partial \rho}{\partial t} \right] + \rho \sum_{\alpha} h_{s,\alpha} \frac{\partial Y_{\alpha}}{\partial t} + h_s \frac{\partial \rho}{\partial t} \\
&= \rho c_p T \left[ \frac{1}{\bar{p}} \frac{\partial \bar{p}}{\partial t} - \sum_{\alpha} \frac{\bar{W}}{W_{\alpha}} \frac{\partial Y_{\alpha}}{\partial t} - \frac{1}{\rho} \frac{\partial \rho}{\partial t} \right] + \rho \sum_{\alpha} h_{s,\alpha} \frac{\partial Y_{\alpha}}{\partial t} + h_s \frac{\partial \rho}{\partial t} \\
&= \frac{\rho c_p T}{\bar{p}} \frac{\partial \bar{p}}{\partial t} + \rho \sum_{\alpha} \left( h_{s,\alpha} - c_p T \frac{\bar{W}}{W_{\alpha}} \right) \frac{\partial Y_{\alpha}}{\partial t} + (h_s - c_p T) \frac{\partial \rho}{\partial t}
\end{aligned} \tag{B.21}$$

The time derivative of the mass fractions, which originates from the species transport equation, is:

$$\frac{\partial Y_{\alpha}}{\partial t} = \frac{1}{\rho} \left[ \nabla \cdot (\rho D_{\alpha} \nabla Y_{\alpha}) + \dot{m}_{\alpha}''' - Y_{\alpha} \frac{\partial \rho}{\partial t} - \nabla \cdot (\rho Y_{\alpha} \mathbf{u}) \right] \tag{B.22}$$

Using (B.22) in (B.21) and summing over species to eliminate the density time derivative we obtain

$$\frac{\partial(\rho h_s)}{\partial t} = \frac{\rho c_p T}{\bar{p}} \frac{\partial \bar{p}}{\partial t} + \sum_{\alpha} \left( h_{s,\alpha} - c_p T \frac{\bar{W}}{W_{\alpha}} \right) \left[ \nabla \cdot (\rho D_{\alpha} \nabla Y_{\alpha}) + \dot{m}_{\alpha}''' - \mathbf{u} \cdot \nabla(\rho Y_{\alpha}) - \rho Y_{\alpha} \nabla \cdot \mathbf{u} \right] \tag{B.23}$$

Plugging (B.23) into (B.8) yields (almost done)

$$\begin{aligned}
\nabla \cdot \mathbf{u} &= \frac{1}{\rho h_s} \frac{D \bar{p}}{Dt} - \frac{c_p T}{h_s} \frac{1}{\bar{p}} \frac{\partial \bar{p}}{\partial t} \\
&\quad + \frac{1}{\rho h_s} [\dot{q}''' - \nabla \cdot \dot{\mathbf{q}}'' - \mathbf{u} \cdot \nabla(\rho h_s)] \\
&\quad + \frac{1}{\rho h_s} \sum_{\alpha} \left( c_p T \frac{\bar{W}}{W_{\alpha}} - h_{s,\alpha} \right) \left[ \nabla \cdot (\rho D_{\alpha} \nabla Y_{\alpha}) - \mathbf{u} \cdot \nabla(\rho Y_{\alpha}) - \rho Y_{\alpha} \nabla \cdot \mathbf{u} \right] \\
&\quad + \frac{1}{\rho h_s} \sum_{\alpha} \left( c_p T \frac{\bar{W}}{W_{\alpha}} - h_{s,\alpha} \right) \dot{m}_{\alpha}''' 
\end{aligned} \tag{B.24}$$

$$\begin{aligned}
\nabla \cdot \mathbf{u} + \frac{1}{\rho h_s} \sum_{\alpha} \left( c_p T \frac{\bar{W}}{W_{\alpha}} - h_{s,\alpha} \right) \rho Y_{\alpha} \nabla \cdot \mathbf{u} &= \frac{1}{\rho h_s} \frac{D \bar{p}}{Dt} - \frac{c_p T}{h_s} \frac{1}{\bar{p}} \frac{\partial \bar{p}}{\partial t} \\
&\quad + \frac{1}{\rho h_s} [\dot{q}''' - \nabla \cdot \dot{\mathbf{q}}'' - \mathbf{u} \cdot \nabla(\rho h_s)] \\
&\quad + \frac{1}{\rho h_s} \sum_{\alpha} \left( c_p T \frac{\bar{W}}{W_{\alpha}} - h_{s,\alpha} \right) \left[ \nabla \cdot (\rho D_{\alpha} \nabla Y_{\alpha}) - \mathbf{u} \cdot \nabla(\rho Y_{\alpha}) + \dot{m}_{\alpha}''' \right]
\end{aligned} \tag{B.25}$$

$$\begin{aligned}
\nabla \cdot \mathbf{u} + \left( \frac{c_p T}{h_s} - 1 \right) \nabla \cdot \mathbf{u} &= \frac{1}{\rho h_s} \frac{D \bar{p}}{Dt} - \frac{c_p T}{h_s} \frac{1}{\bar{p}} \frac{\partial \bar{p}}{\partial t} \\
&\quad + \frac{1}{\rho h_s} [\dot{q}''' - \nabla \cdot \dot{\mathbf{q}}'' - \mathbf{u} \cdot \nabla(\rho h_s)] \\
&\quad + \frac{1}{\rho h_s} \sum_{\alpha} \left( c_p T \frac{\bar{W}}{W_{\alpha}} - h_{s,\alpha} \right) \left[ \nabla \cdot (\rho D_{\alpha} \nabla Y_{\alpha}) - \mathbf{u} \cdot \nabla(\rho Y_{\alpha}) + \dot{m}_{\alpha}''' \right]
\end{aligned} \tag{B.26}$$

Finally... (compare with (B.9))

$$\begin{aligned}\nabla \cdot \mathbf{u} &= \frac{1}{\rho c_p T} \frac{\mathrm{D} \bar{p}}{\mathrm{D} t} - \frac{1}{\bar{p}} \frac{\partial \bar{p}}{\partial t} \\ &\quad + \frac{1}{\rho c_p T} [\dot{q}''' - \nabla \cdot \dot{\mathbf{q}}'' - \mathbf{u} \cdot \nabla (\rho h_s)] \\ &\quad + \frac{1}{\rho} \sum_{\alpha} \left( \frac{\bar{W}}{W_{\alpha}} - \frac{h_{s,\alpha}}{c_p T} \right) \left[ \nabla \cdot (\rho D_{\alpha} \nabla Y_{\alpha}) - \mathbf{u} \cdot \nabla (\rho Y_{\alpha}) + \dot{m}_{\alpha}''' \right]\end{aligned}\quad (\text{B.27})$$

## B.4 Equivalence between Divergence Expressions

The equivalence between (B.7) and (B.9) is apparent based on the following:

$$\begin{aligned}-\frac{1}{\rho c_p T} \mathbf{u} \cdot \nabla (\rho h_s) - \frac{1}{\rho} \sum_{\alpha} \left( \frac{\bar{W}}{W_{\alpha}} - \frac{h_{s,\alpha}}{c_p T} \right) \mathbf{u} \cdot \nabla (\rho Y_{\alpha}) \\ &= -\frac{1}{\rho c_p T} [\rho \mathbf{u} \cdot \nabla h_s + h_s \mathbf{u} \cdot \nabla \rho] - \frac{1}{\rho} \sum_{\alpha} \left( \frac{\bar{W}}{W_{\alpha}} - \frac{h_{s,\alpha}}{c_p T} \right) [\rho \mathbf{u} \cdot \nabla Y_{\alpha} + Y_{\alpha} \mathbf{u} \cdot \nabla \rho] \\ &= -\frac{1}{c_p T} \mathbf{u} \cdot \sum_{\alpha} [Y_{\alpha} \nabla h_{s,\alpha} + h_{s,\alpha} \nabla Y_{\alpha}] - \frac{1}{\rho} \mathbf{u} \cdot \nabla \rho - \sum_{\alpha} \left( \frac{\bar{W}}{W_{\alpha}} - \frac{h_{s,\alpha}}{c_p T} \right) \mathbf{u} \cdot \nabla Y_{\alpha} \\ &= -\frac{1}{c_p T} \mathbf{u} \cdot \sum_{\alpha} Y_{\alpha} \nabla h_{s,\alpha} - \frac{1}{\rho} \mathbf{u} \cdot \nabla \rho - \sum_{\alpha} \frac{\bar{W}}{W_{\alpha}} \mathbf{u} \cdot \nabla Y_{\alpha} \\ &= -\frac{1}{c_p T} \mathbf{u} \cdot \sum_{\alpha} Y_{\alpha} c_{p,\alpha} \nabla T - \frac{1}{\rho} \mathbf{u} \cdot \nabla \rho - \mathbf{u} \cdot \sum_{\alpha} \bar{W} \nabla (Y_{\alpha} / W_{\alpha}) \\ &= -\mathbf{u} \cdot \left[ \frac{1}{T} \nabla T + \frac{1}{\rho} \nabla \rho + \bar{W} \nabla (1/\bar{W}) \right] \\ &= -\frac{1}{\bar{p}} \mathbf{u} \cdot \nabla \bar{p}\end{aligned}\quad (\text{B.28})$$

## B.5 Simplifications for Constant Specific Heat

Recall that for an ideal gas we may write

$$c_{p,\alpha} = c_{v,\alpha} + R/W_{\alpha} = \frac{R}{W_{\alpha}} \left( \frac{\gamma_{\alpha}}{\gamma_{\alpha} - 1} \right), \quad (\text{B.29})$$

where  $c_{p,\alpha}$  is the specific heat of  $\alpha$  at constant pressure,  $c_{v,\alpha}$  is the specific heat at constant volume, and  $\gamma_{\alpha} = c_{p,\alpha}/c_{v,\alpha}$ . Commonly, the ratio of specific heats is approximated to be constant, and for fire calculations typically the value for air is chosen,  $\gamma \approx 1.4$ . If we take the reference temperature to be zero, the sensible enthalpy in this case becomes

$$\rho h_s = \rho c_p T = \rho T \sum_{\alpha} Y_{\alpha} c_{p,\alpha} = \rho RT \left( \frac{\gamma}{\gamma - 1} \right) \sum_{\alpha} \frac{Y_{\alpha}}{W_{\alpha}} = \rho \frac{RT}{\bar{W}} \left( \frac{\gamma}{\gamma - 1} \right) = \bar{p} \left( \frac{\gamma}{\gamma - 1} \right) \quad (\text{B.30})$$

Therefore, if  $\bar{p}$  is constant and uniform then  $\rho h_s$  is constant and uniform. Consequently,  $\partial(\rho h_s)/\partial t = 0$  and  $\nabla(\rho h_s) = 0$ , so we require no corrections to the divergence expression. This improves the speed of the code since these divergence corrections are rather expensive. To employ this simplification, the user enters *both* CONSTANT\_SPECIFIC\_HEAT\_RATIO=.TRUE. and STRATIFICATION=.FALSE. on the MISC line of the input file.

## Appendix C

# Gas Phase Absorption Coefficients

The instantaneous absorption coefficient  $\kappa_n$  of the gas mixture at location  $\mathbf{x}$  is calculated as a sum (superposition) of individual gases' gray or band-mean absorption coefficients  $\kappa_{n,i}$

$$\kappa_n(\mathbf{x}) = \sum_i \kappa_{n,i}(Y_i(\mathbf{x}), T(\mathbf{x})) \quad (\text{C.1})$$

For the calculation of  $\kappa_{n,i}(Y_i, T)$ , a narrow-band model, RADCAL [17], has been implemented in FDS. During the initialization stage, RADCAL is used to generate tables of mean absorption coefficients for each individual species in mass fraction range  $0 \leq Y_i \leq 1$  and temperature range  $270 \text{ K} \leq T \leq 2470 \text{ K}$ . During the simulation, the local value of  $\kappa_n$  is found from the nearest temperature value (lookup) and by interpolating the concentration value.

RADCAL computes the spectral properties of the radiation participating species at discrete values of the spectrum (expressed either in wavenumber  $\omega$  or in wavelength  $\lambda$ ) and temperature, and returns two alternative mean absorption coefficients for each spectral band,  $n$ . The first coefficient is the Planck mean coefficient [134]

$$\kappa_{n,i,P}(Y_i, T) = \frac{\pi}{\sigma T^4} \int_{\lambda_{\min}}^{\lambda_{\max}} I_{b,\lambda}(T) 100 \bar{\kappa}_i(\lambda, T) P_i \, d\lambda \quad (\text{C.2})$$

where  $\lambda$  is the wavelength, expressed in units of  $\mu\text{m}$ ,  $P_i = Y_i / (Y_i + (1 - Y_i)W_{N_2})$  is the partial pressure of participating species  $i$ , in units of atm, and  $\bar{\kappa}_i$  is the spectral absorption coefficient of participating species  $i$ , in units of  $\text{atm}^{-1}\text{cm}^{-1}$ . Note that the temperature used in the calculation of  $\kappa_n$  is the local gas temperature; thus,  $\kappa_{n,i,P}(P, T)$  is a function of the gas phase temperature and partial pressure and is independent of the pathlength. Its units are  $1/\text{m}$ . The factor 100 is introduced to convert  $\bar{\kappa}_i$  from  $\text{atm}^{-1}\text{cm}^{-1}$  to  $\text{atm}^{-1}\text{m}^{-1}$ . RADCAL can do these calculations for mixtures of several species, but in FDS, it is used for a single species at the time.

The source term  $I_{b,\lambda}(T)$  is the *Planck blackbody distribution law* which expresses the equilibrium rate of radiant energy emitted from a blackbody at temperature,  $T$ , and as function of wavelength,  $\lambda$ . Formally, the monochromatic blackbody radiant energy emitted at a wavelength  $\lambda$  is given by [135]

$$I_{b,\lambda}(T) \, d\lambda = \frac{2hc^2 \lambda^{-5}}{\exp\left(\frac{hc}{k_B \lambda T}\right) - 1} \, d\lambda \quad (\text{C.3})$$

Here,  $h$  is the Planck constant ( $6.626 \times 10^{-34} \text{ J s}$ ),  $c$  is the speed of light in vacuum ( $2.998 \times 10^8 \text{ m/s}$ ), and  $k_B$  is the Boltzmann constant ( $1.381 \times 10^{-23} \text{ J/K}$ ) [136].  $I_{b,\lambda}(T)$  is in units of  $\text{W/m}^2/\text{str}/\text{m}$ ; the wavelengths are converted from  $\mu\text{m}$  to  $\text{m}$ .

The second coefficient is the so-called *path mean* or *effective* absorption coefficient,  $\kappa_{n,i,e}(Y_i, T, L)$  which is defined according to the following equation

$$\underbrace{\int_{\lambda_{\min}}^{\lambda_{\max}} I(\lambda, L, Y_i, T, T_{\text{rad}}) d\lambda}_{I_{\text{tot}}} = \frac{\sigma}{\pi} [(1 - e^{-\kappa_{n,i,e} L}) T^4 + e^{-\kappa_{n,i,e} L} T_{\text{rad}}^4] \quad (\text{C.4})$$

where  $L$  is the path length and  $T_{\text{rad}}$  is the effective temperature of flame radiation. RADCAL calculates the left hand integral by calculating the intensity leaving a uniform gas layer of equivalent thickness  $L$ , bounded by a black wall at temperature  $T_{\text{rad}}$ , for a large number of narrow spectral bands. Solving from equation C.4 gives

$$\kappa_{n,i,e}(Y_i, T, L) = -\frac{1}{L} \ln \left( \frac{I_{\text{tot}} - I_b(T)}{I_b(T_{\text{rad}}) - I_b(T)} \right) \quad (\text{C.5})$$

where  $I_b(T) = \sigma T^4 / \pi$ .

By default in FDS, the pathlength,  $L$ , is 10 cm. It can also be specified by the user. If  $T = T_{\text{rad}}$  the intensity does not depend on  $\kappa_{n,i,e}$ , and the value  $\kappa_{n,i,e}(T_{\text{rad}})$  is therefore interpolated from the neighboring temperatures.

In cases with only one band ( $N=1$ ), the smaller of the two absorption coefficients is used:

$$\kappa_{n,i} = \min \left( \kappa_{n,i,p}(Y_i, T), \kappa_{n,i,e}(Y_i, T, L) \right) \quad (\text{C.6})$$

If  $N > 1$  or  $L = 0$ ,  $\kappa_{n,i} = \kappa_{n,i,p}$ . Note that the spectral data within RADCAL are used whenever the gas mixture contains water vapor, fuel or combustion products, regardless of the number of radiation bands  $N$ .

**Note on wavenumber, wavelength, and frequency:** some confusion might arise when dealing with the various quantities describing the wave nature of radiation. These quantities are wavenumber  $\omega$ , wavelength  $\lambda$ , and frequency denoted here  $v$ . Most users may be familiar with the frequency  $v$ , in units of *hertz*, Hz, representing the number of cycles per second. While this unit is preferred for radiation waves of low energy such as radio waves, wavelength and wavenumber are preferred for waves of higher energy. Wavenumber and wavelength are related to frequency through [135]

$$\lambda = c/v \text{ and } \omega = v/c \quad (\text{C.7})$$

where  $c$  is the speed of light in a vacuum. The wavelength,  $\lambda$ , represents the distance traveled by the wave during one cycle, assuming it travels at the speed of light in a vacuum. Its units are commonly expressed in  $\mu\text{m}$ . Wavenumber,  $\omega$ , is the reciprocal of the wavelength. It represents the number of cycles per unit length. In most infrared spectroscopic work, it is expressed in units of  $\text{cm}^{-1}$ . This is the unit used in the sections below. One can easily switch from wavenumber in units of  $\text{cm}^{-1}$  to wavelength in units of  $\mu\text{m}$  using the relation

$$\lambda \mu\text{m} = 10000/\omega \text{ cm}^{-1} \quad (\text{C.8})$$

Finally, the user who wishes to express the Planck blackbody distribution law as a function of wavenumber should take caution when performing the change of variables. One should start by expressing that the radiant energy emitted at a wavelength  $\lambda$  is the same as the radiant energy emitted at the corresponding wavenumber  $\omega$  [134]

$$I_{b,\lambda}(T) d\lambda = -I_{b,\omega}(T) d\omega \quad (\text{C.9})$$

the negative sign is introduced because  $\omega$  is the reciprocal of  $\lambda$ . Since  $\lambda = 1/\omega$ , it comes

$$\frac{d\lambda}{d\omega} = -\frac{1}{\omega^2} \quad (\text{C.10})$$

Equation C.9, can be rewritten using Eq. C.3 as

$$I_{b,\omega}(T) d\omega = \frac{2hc^2\omega^3}{\exp\left(\frac{hc\omega}{k_B T}\right) - 1} d\omega \quad (\text{C.11})$$

$I_{b,\omega}(T)$  is in units of  $\text{W}/\text{m}/\text{str}/\text{m}^{-1}$ ; the wavenumbers are converted from  $\text{cm}^{-1}$  to  $\text{m}^{-1}$ . The user who wishes to analyze the fuel bands presented below (given in wavenumber) with the Planck blackbody distribution law should use Eq. C.11 but should NOT use Eq. C.3 with  $\lambda = 1/\omega$ .

A combination of molecular models and data tables are used to compute the spectral radiative properties of the radiation participating species. The original version of RADCAL includes spectral properties of  $\text{CO}_2$ ,  $\text{H}_2\text{O}$ ,  $\text{CO}$ , and  $\text{CH}_4$  that are either modeled through quantitative molecular spectroscopy derivations or tabulated from the fitting of experimental data into appropriate statistical narrow band models [17]. The original RADCAL data have been supplemented with new tabulated experimental data for the following fuels:

- Ethylene:  $\text{C}_2\text{H}_4$
- Ethane:  $\text{C}_2\text{H}_6$
- Propylene:  $\text{C}_3\text{H}_6$
- Propane:  $\text{C}_3\text{H}_8$
- Toluene:  $\text{C}_7\text{H}_8$
- *n*-Heptane:  $\text{C}_7\text{H}_{16}$
- Methanol:  $\text{CH}_3\text{OH}$
- Methyl Methacrylate:  $\text{C}_5\text{H}_8\text{O}_2$

These new data have been obtained through FTIR measurements for wavenumbers between  $700 \text{ cm}^{-1}$  and  $4000 \text{ cm}^{-1}$  [137]. A useful quantity to compare the relative importance of the different IR bands is provided by the integrated band intensity,  $\alpha_i$ , defined for the  $i$ th participating species as [138]:

$$\alpha_i(T) = \int_{\omega_{\min}}^{\omega_{\max}} \bar{\kappa}_i(\omega', T) d\omega' \quad (\text{C.12})$$

whose units are  $\text{atm}^{-1}\text{cm}^{-2}$ . The value of the spectral absorption coefficient,  $\bar{\kappa}_i$ , is averaged over a narrow band whose spectral width,  $\Delta\omega$ , varies from  $5 \text{ cm}^{-1}$  for  $\omega < 1100 \text{ cm}^{-1}$ , to  $25 \text{ cm}^{-1}$  for  $1100 \text{ cm}^{-1} \leq \omega < 5000 \text{ cm}^{-1}$ , and to  $50 \text{ cm}^{-1}$  for  $5000 \text{ cm}^{-1} \leq \omega$ .

The subsections below briefly describe the molecular bands where the species are active for each of the gas-phase radiative species, and provide for most of them the integrated band intensity of their most important bands at the indicated temperature. Outside these bands, the species are transparent. At the start of a simulation, the absorption coefficients are calculated using RADCAL and then tabulated as a function of species concentration and temperature. During the simulation, the local absorption coefficient is interpolated from the table of values. The contributions of individual species are summed to the total absorption coefficient.

Table C.1: Spectral bands of CO<sub>2</sub> included in RADCAL.

Band #	Bounds (cm <sup>-1</sup> )		Method
1	500	880	tabulated
2	880	1100	modeled
3	1975	2475	modeled
4	3050	3800	modeled
5	4550	5275	modeled

### Carbon Dioxide: CO<sub>2</sub>

Carbon dioxide is a linear molecule and has four vibrational modes, but only two fundamental IR vibration frequencies [139]. It has five distinct bands that are included in RADCAL, see Table C.1. The strongest band in the CO<sub>2</sub> spectrum is Band 3. At 300 K, it has an integrated band intensity of 2963 atm<sup>-1</sup>cm<sup>-2</sup>. The tabulated data were obtained from experiments with temperatures ranging from 300 K to 2400 K using the Goody statistical narrow band model.

### Carbon Monoxide: CO

Carbon Monoxide is a diatomic molecule and as such, it has only one vibrational mode [139]. RADCAL includes one distinct band, see Table C.2. It corresponds to the stretching of the triple bond C ≡ O. The

Table C.2: Spectral bands of CO included in RADCAL.

Band #	Bounds (cm <sup>-1</sup> )		Method
1	1600	2400	modeled

first overtone (centered at  $\omega \approx 4260$  cm<sup>-1</sup>) is not accounted for; its integrated band intensity is negligible at standard temperature and pressure. At 295 K, the integrated band intensity of Band 1 is 260 atm<sup>-1</sup>cm<sup>-2</sup>. The statistical narrow band model associated with CO is the Goody model. Recommended temperatures of use range from 295 K to 2500 K.

### Water Vapor: H<sub>2</sub>O

Due to the non-linearity of its molecular structure, the IR spectrum of water vapor is complex and broad [139]. In RADCAL, water vapor spectrum from 50 cm<sup>-1</sup> to 9300 cm<sup>-1</sup> is considered. Data in RADCAL are provided by Ludwig *et al.* [58]. Experimental data have been fitted using the statistical Goody narrow band model. The strongest bands at standard temperature and pressure are located in the ranges [50 – 2100] cm<sup>-1</sup>:  $\alpha = 300$  atm<sup>-1</sup>cm<sup>-2</sup>, and [3000 – 4000] cm<sup>-1</sup>:  $\alpha = 220$  atm<sup>-1</sup>cm<sup>-2</sup>.

### Methane: CH<sub>4</sub>

Methane is a spherical top molecule of tetrahedral shape with the carbon atom occupying the center of the tetrahedron [139]. It belongs to the point group  $T_d$ . The methane IR spectrum is the result of the vibration-rotation modes of the C – H groups. It has nine vibrational modes, but due to its symmetry, this translates into only two distinct IR active fundamental vibration frequencies. In RADCAL, the methane IR spectrum is divided into three distinct bands (fundamentals + degenerates), see Table C.3. The strongest

Table C.3: Spectral bands of CH<sub>4</sub> included in RADCAL.

Band #	Bounds (cm <sup>-1</sup> )		Method	Assignment	$\alpha(T = 296 \text{ K}) (\text{atm}^{-1}\text{cm}^{-2})$
1	1150	1600	tabulated	C – H Bend	237
2	2700	3250	tabulated	C – H Stretch	212
3	3400	5000	modeled	C – H Stretch	

bands are Bands 1 and 2 which at standard temperature and pressure have an integrated band intensity of 237 atm<sup>-1</sup>cm<sup>-2</sup> and 212 atm<sup>-1</sup>cm<sup>-2</sup>, respectively. The tabulated data were obtained from high resolution FTIR experiments with temperatures varying from 300 K to 1400 K [137]. The spectral absorption coefficients were obtained assuming the FTIR measurements to be in the weak line regime and applying the Beer-Lambert Law to the experimental spectral transmissivity.

#### Ethylene: C<sub>2</sub>H<sub>4</sub>

Ethylene is a molecule with a plane symmetrical form and belongs to the point group  $D_{2h}$  [139]. The ethylene IR spectrum is the result of the vibration-rotation modes of the C = C, CH, and CH<sub>2</sub> groups. It has 12 vibrational modes. In RADCAL, its IR spectrum is divided into four distinct bands, see Table C.4. Band 1 is the strongest absorbing band. All the ethylene IR spectral absorption data were obtained from high

Table C.4: Spectral bands of C<sub>2</sub>H<sub>4</sub> included in RADCAL.

Band #	Bounds (cm <sup>-1</sup> )		Method	Assignment	$\alpha(T = 296 \text{ K}) (\text{atm}^{-1}\text{cm}^{-2})$
1	780	1250	tabulated	CH <sub>2</sub> Bend	366
2	1300	1600	tabulated	CH <sub>2</sub> Bend	43
3	1750	2075	tabulated	C = C Stretch	20
4	2800	3400	tabulated	C – H Stretch	183

resolution FTIR experiments with temperatures varying from 296 K to 801 K [137]. The spectral absorption coefficients were obtained by fitting the experimental spectral transmissivity of a homogeneous column of isothermal ethylene with a total pressure of 1 atm using the Goody model.

#### Ethane: C<sub>2</sub>H<sub>6</sub>

Ethane has a three-fold axis of symmetry and belongs to the point group  $D_{3d}$  [139]. The ethane IR spectrum is the result of the vibration-rotation modes of the C – C, CH, and CH<sub>2</sub> groups. It has 18 vibrational modes; its IR spectrum is divided into three distinct bands, see Table C.5. Band 3 corresponds to the stretching of

Table C.5: Spectral bands of C<sub>2</sub>H<sub>6</sub> included in RADCAL.

Band #	Bounds (cm <sup>-1</sup> )		Method	Assignment	$\alpha(T = 296 \text{ K}) (\text{atm}^{-1}\text{cm}^{-2})$
1	730	1095	tabulated	CH <sub>3</sub> Rock	29
2	1250	1700	tabulated	CH Bend	64
3	2550	3375	tabulated	CH Stretch	761

CH and is the strongest absorbing band. At standard temperature and pressure, its integrated band intensity

is more than 10 times the value of Band 2, and more than 20 times the value of Band 1. All the ethane IR spectral absorption data were obtained from high resolution FTIR experiments with temperatures varying from 296 K to 1000 K [137]. The spectral absorption coefficients were obtained by fitting the experimental spectral transmissivity of a homogeneous column of isothermal ethane with a total pressure of 1 atm using the Elsasser model.

### Propylene: C<sub>3</sub>H<sub>6</sub>

Propylene has only one plane of symmetry and belongs to the point group  $C_s$  [139]. The propylene IR spectrum is the result of the vibration-rotation modes of the C – C, C = C, CH, CH<sub>2</sub>, and CH<sub>3</sub> groups. It has 21 vibrational modes; its IR spectrum is divided into three distinct bands, see Table C.6. Band 3

Table C.6: Spectral bands of C<sub>3</sub>H<sub>6</sub> included in RADCAL.

Band #	Bounds (cm <sup>-1</sup> )	Method	Assignment	$\alpha(T = 296 \text{ K}) (\text{atm}^{-1}\text{cm}^{-2})$
1	775	1150	tabulated	C – C Stretch, CH <sub>3</sub> Rock
2	1225	1975	tabulated	C = C Stretch, CH Bend
3	2650	3275	tabulated	CH Stretch

corresponds to the stretching of CH and is the strongest of all propylene absorbing bands. All the propylene IR spectral absorption data were obtained from high resolution FTIR experiments with temperatures varying from 296 K to 1003 K [137]. The spectral absorption coefficients were obtained by fitting the experimental spectral transmissivity of a homogeneous column of isothermal propylene with a total pressure of 1 atm using the Goody model.

### Propane: C<sub>3</sub>H<sub>8</sub>

Propane has two planes of symmetry and two axes of rotation. It belongs to the point group  $C_{2v}$  [139]. The propane IR spectrum is the result of the vibration-rotation modes of the C – C, CH<sub>2</sub>, CH<sub>3</sub> groups. It has 27 vibrational modes; its IR spectrum is divided into two distinct bands, see Table C.7. Band 2 corresponds to

Table C.7: Spectral bands of C<sub>3</sub>H<sub>8</sub> included in RADCAL.

Band #	Bounds (cm <sup>-1</sup> )	Method	Assignment	$\alpha(T = 295 \text{ K}) (\text{atm}^{-1}\text{cm}^{-2})$
1	1175	1675	tabulated	CH <sub>3</sub> Bending
2	2550	3375	tabulated	CH <sub>3</sub> , CH <sub>2</sub> Stretch

the stretching of CH<sub>3</sub> and CH<sub>2</sub> and is the strongest of all propane absorbing bands. For similar conditions, Band 1 has a much lower integrated band intensity. All the propane IR spectral absorption data were obtained from high resolution FTIR experiments with temperatures varying from 295 K to 1009 K [137]. The spectral absorption coefficients were obtained by fitting the experimental spectral transmissivity of a homogeneous column of isothermal propane with a total pressure of 1 atm using the Goody model.

### Toluene: C<sub>7</sub>H<sub>8</sub>

Toluene has only one plane of symmetry. It belongs to the point group  $C_s$  [140]. The toluene IR spectrum is the result of the vibration-rotation modes of the C = C, CH, and CH<sub>3</sub> groups. It has 39 vibrational modes.

For ease of modeling using statistical narrow band models, its IR spectrum has been divided into five distinct bands, see Table C.8. Band 5 corresponds to the stretching of CH<sub>3</sub> and CH, and it is the strongest absorbing

Table C.8: Spectral bands of C<sub>7</sub>H<sub>8</sub> included in RADCAL.

Band #	Bounds (cm <sup>-1</sup> )		Method	Assignment	$\alpha(T = 300 \text{ K}) (\text{atm}^{-1}\text{cm}^{-2})$
1	700	805	tabulated	CH Bending	237
2	975	1175	tabulated	CH Bending	40
3	1275	1650	tabulated	CH <sub>3</sub> Bending	166
4	1650	2075	tabulated	C = C Stretching	101
5	2675	3225	tabulated	CH <sub>3</sub> , CH Stretching	510

band. All the toluene IR spectral absorption data were obtained from high resolution FTIR experiments with temperatures varying from 300 K to 795 K [137]. The spectral absorption coefficients were obtained by fitting the experimental spectral transmissivity of a homogeneous column of isothermal toluene with a total pressure of 1 atm using the Goody model.

### **n-Heptane:** C<sub>7</sub>H<sub>16</sub>

*n*-heptane has two planes of symmetry and two axes of rotation. It belongs to the point group C<sub>2v</sub> [140]. The *n*-heptane IR spectrum results from the vibration-rotation modes of the C – C, CH<sub>2</sub>, and CH<sub>3</sub> groups. It has 63 vibrational modes. For ease of modeling using statistical narrow band models, its IR spectrum has been divided into two distinct bands, see Table C.9. Band 2 corresponds to the stretching of CH<sub>3</sub> and CH<sub>2</sub>

Table C.9: Spectral bands of C<sub>7</sub>H<sub>16</sub> included in RADCAL.

Band #	Bounds (cm <sup>-1</sup> )		Method	Assignment	$\alpha(T = 293 \text{ K}) (\text{atm}^{-1}\text{cm}^{-2})$
1	1100	1800	tabulated	CH <sub>2</sub> , CH <sub>3</sub> Bending	298
2	2250	3275	tabulated	CH <sub>2</sub> , CH <sub>3</sub> Stretching	3055

groups, and is the strongest absorbing band. All the *n*-heptane IR spectral absorption data were obtained from high resolution FTIR experiments with temperatures varying from 293 K to 794 K [137]. The spectral absorption coefficients were obtained by fitting the experimental spectral transmissivity of a homogeneous column of isothermal *n*-heptane with a total pressure of 1 atm using the Goody model.

### **Methanol:** CH<sub>3</sub>OH

Methanol has only one plane of symmetry. It belongs to the point group C<sub>s</sub> [139]. The methanol IR spectrum results from the vibration-rotation modes of the C – O, OH, and CH<sub>3</sub> groups. It has 12 vibrational modes. For ease of modeling using statistical narrow band models, its IR spectrum has been divided into four distinct bands, see Table C.10. Band 3 corresponds to the stretching of the CH<sub>3</sub> group and is the strongest absorbing band. All the methanol IR spectral absorption data were obtained from high resolution FTIR experiments with temperatures varying from 293 K to 804 K [137]. The spectral absorption coefficients were obtained by fitting the experimental spectral transmissivity of a homogeneous column of isothermal methanol with a total pressure of 1 atm using the Goody model.

Table C.10: Spectral bands of CH<sub>3</sub>OH included in RADCAL.

Band #	Bounds (cm <sup>-1</sup> )		Method	Assignment	$\alpha(T = 293 \text{ K}) (\text{atm}^{-1}\text{cm}^{-2})$
1	825	1125	tabulated	C – O Stretching	593
2	1125	1700	tabulated	CH <sub>3</sub> , OH Bending	197
3	2600	3225	tabulated	CH <sub>3</sub> Stretching	684
4	3525	3850	tabulated	OH Stretching	112

### Methyl Methacrylate: C<sub>5</sub>H<sub>8</sub>O<sub>2</sub>

Methyl Methacrylate or MMA has the most complex IR spectrum of all the fuels presented above. With 15 atoms, it has 39 vibrational modes [140]. The MMA IR spectrum results from the vibration-rotation modes of the C – O, C = O, C = C, CH<sub>2</sub>, and CH<sub>3</sub> groups. For ease of modeling using statistical narrow band models, its IR spectrum has been divided into six distinct bands, see Table C.11. Band 3 corresponds to

Table C.11: Spectral bands of C<sub>5</sub>H<sub>8</sub>O<sub>2</sub> included in RADCAL.

Band #	Bounds (cm <sup>-1</sup> )		Method	Assignment	$\alpha(T = 396 \text{ K}) (\text{atm}^{-1}\text{cm}^{-2})$
1	750	875	tabulated	CH <sub>2</sub> Bending	42
2	875	1050	tabulated	CH <sub>2</sub> Bending	131
3	1050	1250	tabulated	C – O Stretching	800
4	1250	1550	tabulated	CH <sub>3</sub> Bending	490
5	1550	1975	tabulated	C = C, C = O Stretching	538
6	2650	3275	tabulated	CH <sub>2</sub> , CH <sub>3</sub> Stretching	294

the stretching of the C – O group and has the highest integrated band intensity. All the MMA IR spectral absorption data were obtained from high resolution FTIR experiments with temperatures varying from 396 K to 803 K [137]. The spectral absorption coefficients were obtained by fitting the experimental spectral transmissivity of a homogeneous column of isothermal MMA with a total pressure of 1 atm using the Goody model.

### Statistical Narrow Band Models

This section briefly describes the statistical models used to obtain most of the tabulated species IR spectral absorption coefficients at different temperatures,  $\bar{\kappa}_i(\omega, T)$ . Narrow band models are used in lieu of line-by-line models to represent the IR spectra of radiating species in engineering applications. In the narrow band approach, the whole spectrum is divided into small spectral bands (typically several cm<sup>-1</sup>), and different statistical approaches are used to compute the average radiative properties over these narrow bands. Two main models are presented below: the Elsasser model and the Goody model. Both models assume Lorentz lines.

The Elsasser model assumes all the lines to have the same shape, same strength, and to be equally spaced from each other. In this model, the spectral transmissivity,  $\bar{\tau}_\omega$ , of a homogeneous isothermal column filled with only some gas of species *i*, at a total pressure  $P_T$  and with an optical pathlength  $U = P_i L$  (*L* being the column physical length and  $P_i$  the *i*th participating radiating species partial pressure), is given by the

expression [141]:

$$\bar{\tau}_\omega = 1 - \operatorname{erf} \left( \frac{\sqrt{\pi} \bar{\kappa}_i U}{\sqrt{1 + \frac{\pi \bar{\kappa}_i U}{4 \bar{\gamma}_i P_T}}} \right) \quad (\text{C.13})$$

where  $\bar{\gamma}_i$  is the spectral fine structure parameter of the narrow band. Its units are in  $\text{atm}^{-1}$ .  $\bar{\kappa}_i$  is the spectral absorption coefficient of the narrow band. Its units are in  $\text{atm}^{-1}\text{cm}^{-1}$ . The Goody model assumes all the lines to have the same shape, but to be randomly spaced within the narrow band, and their line strength follows an exponential distribution. For this model, the spectral transmissivity is given by the expression [141]:

$$\bar{\tau}_\omega = \exp \left( -\frac{\bar{\kappa}_i U}{\sqrt{1 + \frac{\bar{\kappa}_i U}{4 \bar{\gamma}_i P_T}}} \right) \quad (\text{C.14})$$

For both models, the two narrow band spectral quantities of the  $i$ th species ( $\bar{\kappa}_i$  and  $\bar{\gamma}_i$ ) are obtained either from line-by-line calculations or by fitting experimental data.

**Note:** For all the tabulated data, a linear interpolation of  $\bar{\kappa}_i$  and  $\bar{\gamma}_i$  in temperature and/or in wavenumber is performed by RADCAL when necessary. If the temperature sought is out of the tabulated data range, then the data at the nearest temperature are used.



## Appendix D

# Numerical Methods for Integration of Complex Chemistry

To solve the system of ordinary differential equations used for complex chemistry (e.g., multi-step reactions, reversible reactions), FDS uses a second-order Runge-Kutta (RK2) scheme as the foundation for its numerical integrator. As discussed below, the scheme is augmented with Richardson extrapolation, which increases the accuracy to fourth-order and provides a means of error control. The general procedure of using an explicit integration scheme with an error controller to maintain stability and speed is similar to that of Mott and Oran [142].

Our goal is to integrate the ODE

$$\frac{dY}{dt} = f(Y) \quad (\text{D.1})$$

where  $f(Y)$  represents a reaction rate law and  $Y$  is a species mass fraction (temperature dependence of the rate law is frozen at the initial condition). The total time interval is the FDS time step,  $\delta t$ , and the iteration substep is  $\Delta t$ . The scheme is given by

$$Y^* = Y^k + \Delta t^k f(Y^k) \quad (\text{D.2})$$

$$Y^{k+1} = \frac{1}{2} (Y^k + Y^* + \Delta t^k f(Y^*)) \quad (\text{D.3})$$

Here  $k$  is the iteration index. The number of sub-time intervals is determined by an error controller, as described below. This second-order scheme is an improvement upon the first-order explicit scheme used for simple chemistry. To maintain stability for stiff problems, however, an intractable number of time steps may be required.

### Richardson Extrapolation

To overcome this issue, we use Richardson extrapolation. Richardson extrapolation increases the order of accuracy and provides a means of error control. Suppose you have an exact solution  $A$  represented by a numerical approximation using the interval  $h$  such that the error,  $A - A(h)$ , can be represented by a polynomial expansion of  $h$ . For two different intervals  $h$  and  $h/2$  we may write

$$A_1 = A(h) + \underbrace{a_1(h) + a_2(h)^2 + O(h^3)}_{O(h)} \quad (\text{D.4})$$

$$A_2 = A\left(\frac{h}{2}\right) + \underbrace{a_1\left(\frac{h}{2}\right) + a_2\left(\frac{h}{2}\right)^2 + O(h^3)}_{O(h)} \quad (\text{D.5})$$

Here the subscript on  $A$  represents subdivisions of the time step  $h$ , where  $A_1$  takes  $h/1$  steps,  $A_2$  takes  $h/2$  steps, and so on. The  $a_1$  terms can be eliminated by subtracting Eq. (D.4) from two times Eq. (D.5):

$$A = 2A_2 - A_1 = 2A\left(\frac{h}{2}\right) - A(h) + \underbrace{a_2\left(\frac{h}{2}\right)^2}_{O(h^2)} + O(h^3) \quad (\text{D.6})$$

which results in a solution with a higher order of error.

This same technique is applied to the second-order Runge-Kutta scheme with three sub-intervals:  $h$ ,  $h/2$ , and  $h/4$ . Because the RK2 scheme was initially second-order, the three-step Richardson extrapolation gives a fourth-order solution [143]:

$$A = \frac{4A_4 - A_2}{3} + O(h^4) \quad (\text{D.7})$$

In Eq. (D.7),  $A$  represents the time updated value of the species concentrations in the mixed reactor zone,  $\hat{Y}_\alpha$ , as found in Eq. (5.28).

## Error Control

In addition to increasing the order of the error in the numerical scheme, Richardson extrapolation also provides an estimate of the error value. This estimate is calculated via a Taylor expansion:

$$\text{error} \approx \frac{1}{15} \left( \frac{4A_4 - A_2}{3} - \frac{4A_2 - A_1}{3} \right) \quad (\text{D.8})$$

The time step ( $\Delta t_{\text{new}}$ ) required to maintain the specified/acceptable error ( $\text{err\_tol}$ ) can be calculated by using the local error (error), the current time step ( $\Delta t$ ), and the order of error within the numerical integration scheme, in this case fourth-order:

$$\Delta t_{\text{new}} = \Delta t \left( \frac{\text{err\_tol}}{\text{error}} \right)^{1/4} \quad (\text{D.9})$$

Equation (D.9) indicates that if the error estimate is large relative to the tolerance, then the new time step decreases; whereas if the error is small, then the new time step increases. This dynamic time stepping scheme improves computational efficiency because it allows the integrator to take the largest time step possible bounded by either the error tolerance or the global simulation time step, thus minimizing the total number of integration steps required.

## Maintaining Boundedness

For an arbitrary reaction rate and time step, the forward integration may lead to unbounded species mass fractions. To avoid this issue, we use a method which limits a sub-time step based on the time it takes the limiting species to go out of bounds (see Kahaner et al. [144]). That substep is taken, then the right-hand-sides of the ODEs are recomputed with the new concentrations. This process is repeated until either all reaction rates are zero or the time has reached the total LES time step. So-called “infinitely fast” reactions are treated as second-order reactions (assuming two reactants) with zero activation energy and large Arrhenius constants. This method maintains the proper reactant ratios in the case of multiple infinitely fast reactions.

## A Total Variation Scheme for Detecting Equilibrium

The stiffness associated with a flame front is handled by the error controller and variable time stepping discussed above. The scheme discussed in this section automatically detects chemical equilibrium, the

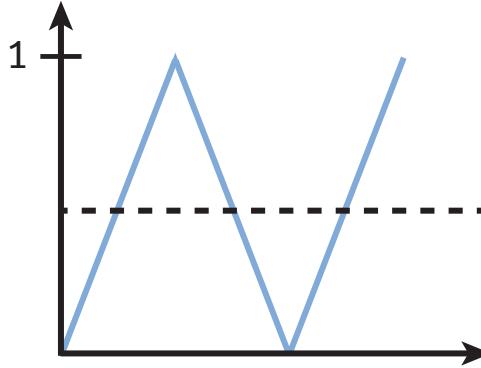


Figure D.1: Illustration of a quantity fluctuating between 0 and 1 about a fixed value.

explicit integration of which is also a stiff problem. Even with error control, the numerical solution may exhibit numerical fluctuations between the bounds of the error tolerance. This is an undesirable condition which often occurs in cases with competing or reversible reactions.

If the values are fluctuating around a fixed value bounded by the error tolerance, the integrator is doing extra work by continuing to calculate every fluctuation within a time step. To minimize the computational expense, a total variation (TV) scheme has been developed. This scheme examines the differences in species mass fractions at four consecutive sub-time steps. Within this four point stencil, there are three differences for each species mass fraction. The TV scheme compares the absolute value of the sum of the differences to the sum of the absolute values of the differences. For a monotonic function, the following condition must hold:

$$\sum_{i=1}^4 |Y(i+1) - Y(i)| = \left| \sum_{i=1}^4 (Y(i+1) - Y(i)) \right| \quad (\text{D.10})$$

The reader unfamiliar with the concept of “data variation” should stop and think about this for a moment. It says that, regardless of other complexities, the monotonicity constraint (no fluctuations) is sufficient to guarantee the condition stated in Eq. (D.10). The corollary is that if Eq. (D.10) does not hold, then we are guaranteed to have some level of fluctuation.

The condition which corresponds to the extreme case of repeating fluctuations between the error tolerance bounds for our four point stencil is

$$\sum_{i=1}^4 |Y(i+1) - Y(i)| = 3 \left| \sum_{i=1}^4 (Y(i+1) - Y(i)) \right| \quad (\text{D.11})$$

This is illustrated in Fig. D.1, which shows a quantity fluctuating between 0 and 1 about a fixed point. We can compare the sum of the absolute value of the differences between the four points,

$$\sum_{i=1}^4 |Y(i+1) - Y(i)| = |1 - 0| + |0 - 1| + |1 - 0| = 3 \quad (\text{D.12})$$

to the absolute value of the sum of the differences,

$$\left| \sum_{i=1}^4 (Y(i+1) - Y(i)) \right| = |(1 - 0) + (0 - 1) + (1 - 0)| = 1 \quad (\text{D.13})$$

The ratio of Eq. (D.12) and Eq. (D.13) is 3, which indicates that this data is fluctuating. For the implementation of this scheme in FDS, instead of comparing directly to 3, we use 2.9 because floating point division might not return a value of exactly 3. If fluctuations have been identified, then the values at the fourth point in the stencil are frozen and become the final values at the end of an integration time step. This prevents unnecessary sub-time steps being taken by the integrator. Because the error controller is bounding the solution, any one of the values within the stencil is also within the error tolerance and is therefore acceptable.

## Appendix E

# The Unmixed Fraction

In Sec. 5.2.2 we present the evolution equation for an important variable, the unmixed fraction,  $\zeta$ , in the batch reactor model for turbulent combustion. This appendix explains the development of that equation.

### Moments of the PDF Transport Equation

The cell mean mass fraction of species  $\alpha$  is denoted  $\tilde{Y}_\alpha(t)$ . A fluid parcel at any point within the cell exists in one of two states: completely unmixed or completely mixed. Let  $\hat{Y}_\alpha(t)$  denote the mass fraction of  $\alpha$  in the mixed reactor zone, initially equal to the cell mean,  $\hat{Y}_\alpha(0) = \tilde{Y}_\alpha^0 \equiv \tilde{Y}_\alpha(0)$ . With  $\psi_\alpha \in [0, 1]$  representing the sample space for the composition, the subgrid probability density function (PDF) may be written as

$$f(\psi_\alpha; t) = w_1 \delta(0 - \psi_\alpha) + w_2 \delta(1 - \psi_\alpha) + w_3 \delta(\hat{Y}_\alpha[t] - \psi_\alpha) \quad (\text{E.1})$$

where  $\delta(x)$  is the Dirac delta function. In other words, if we look at a specific point, the mass fraction of species  $\alpha$  may only be 0, 1, or equal to the mixed zone value,  $\hat{Y}_\alpha$  (see Fig. E.1). The weights  $w_i$  must satisfy integral constraints on the cell:  $\int f(\psi_\alpha; t) d\psi_\alpha = 1$ ,  $\int f(\psi_\alpha; t) \psi_\alpha d\psi_\alpha = \tilde{Y}_\alpha(t)$ .

For convenience, we define the *unmixed fraction*,  $\zeta(t)$ , as the fraction of mass within the cell existing as either 0 or 1. To satisfy the integral constraints, the PDF weights are set to

$$w_1 = \zeta (1 - \tilde{Y}_\alpha^0) \quad (\text{E.2})$$

$$w_2 = \zeta \tilde{Y}_\alpha^0 \quad (\text{E.3})$$

$$w_3 = 1 - \zeta \quad (\text{E.4})$$

As discussed by Pope [15], the PDF (E.1) evolves by the Fokker-Planck equation:

$$\frac{\partial f}{\partial t} = -\frac{\partial}{\partial \psi_\alpha} \left( f \left\langle \frac{\partial Y_\alpha}{\partial t} \middle| \psi_\alpha \right\rangle \right). \quad (\text{E.5})$$

The term on the right in angled brackets is a conditional mean. Here it is modeled using a variant of IEM [49] which we call *interaction by exchange with the mixed mean* or IEMM. When including chemical reaction, the conditional mean is modeled by

$$\left\langle \frac{\partial Y_\alpha}{\partial t} \middle| \psi_\alpha \right\rangle = \frac{1}{\tau_{\text{mix}}} (\hat{Y}_\alpha - \psi_\alpha) + \frac{d\hat{Y}_\alpha}{dt}. \quad (\text{E.6})$$

Using (E.6), the zeroth moment of (E.5) yields (5.19). The zeroth and first moments of (E.5) combine to yield the model for the chemical source term (5.20), once multiplied by density.

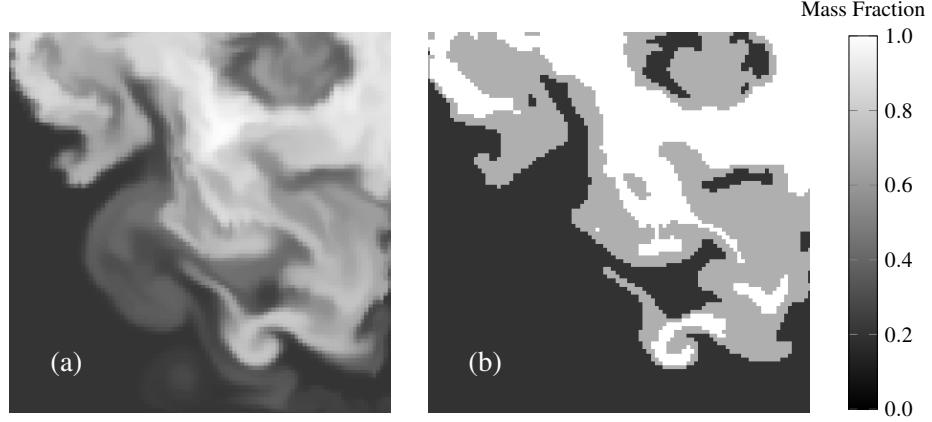


Figure E.1: Subgrid environment at an instant in time in a hypothetical computational cell (batch reactor). (a) Well-resolved scalar field, highly unmixed, after turbulent transport. (b) Idealized subgrid environment: the local mass fraction is either 0, 1, or equal to a mixed mean. In the present model, the gray region is called the “mixed reactor zone” and evolves in time (in volume and composition) during the integration of the batch reactor.

## Derivation of First Moment Equation

$$\int \psi_\beta \left[ \frac{\partial f}{\partial t} = -\frac{\partial}{\partial \psi_\alpha} \left( f \left\langle \frac{\partial Y_\alpha}{\partial t} \middle| \psi_\alpha \right\rangle \right) \right] d\psi_\beta \quad (\text{E.7})$$

LHS:

$$\int \left[ \frac{\partial f \psi_\beta}{\partial t} - f \underbrace{\frac{\partial \psi_\beta}{\partial t}}_0 \right] d\psi_\beta = \frac{d\tilde{Y}_\beta}{dt} \quad (\text{E.8})$$

RHS:

$$\begin{aligned} \int -\psi_\beta \frac{\partial}{\partial \psi_\alpha} \left( f \left\{ \frac{1}{\tau_{\text{mix}}} (\hat{Y}_\alpha - \psi_\alpha) + \frac{d\hat{Y}_\alpha}{dt} \right\} \right) d\psi_\beta = \\ - \underbrace{\int \frac{\partial}{\partial \psi_\alpha} [\psi_\beta f \{ \}] d\psi_\beta}_{0 \text{ by Pope Exercise 12.1 [15]}} + \int f \{ \} \underbrace{\frac{\partial \psi_\beta}{\partial \psi_\alpha}}_{\delta_{\alpha\beta}} d\psi_\beta \end{aligned} \quad (\text{E.9})$$

$$\begin{aligned} &= \frac{1}{\tau_{\text{mix}}} (\hat{Y}_\beta - \tilde{Y}_\beta) + (1 - \zeta) \frac{d\hat{Y}_\beta}{dt} \\ &= \frac{1}{\tau_{\text{mix}}} (\hat{Y}_\beta - [\zeta \tilde{Y}_\beta^0 + (1 - \zeta) \hat{Y}_\beta]) + (1 - \zeta) \frac{d\hat{Y}_\beta}{dt} \\ &= \frac{\zeta}{\tau_{\text{mix}}} (\hat{Y}_\beta - \tilde{Y}_\beta^0) + (1 - \zeta) \frac{d\hat{Y}_\beta}{dt} \end{aligned} \quad (\text{E.10})$$

## Evolution of the Unmixed Fraction

Equation (5.17) may be derived as follows. Differentiating (5.19) in time we get

$$\begin{aligned}\frac{d\tilde{Y}_\alpha}{dt} &= \tilde{Y}_\alpha^0 \frac{d\zeta}{dt} + (1 - \zeta) \frac{d\hat{Y}_\alpha}{dt} - \hat{Y}_\alpha \frac{d\zeta}{dt} \\ &= -\frac{d\zeta}{dt} (\hat{Y}_\alpha - \tilde{Y}_\alpha^0) + (1 - \zeta) \frac{d\hat{Y}_\alpha}{dt}\end{aligned}\quad (\text{E.11})$$

Comparing (E.11) with (E.10) we see that during the reactor step the unmixed fraction evolves by

$$\frac{d\zeta}{dt} = -\frac{\zeta}{\tau_{\text{mix}}}. \quad (\text{E.12})$$

Note that while (5.20) invokes (E.12) in its derivation, (E.10) does not—it is an independent derivation that relies on the choice of mixing model, in this case a variant of IEM as shown in (E.6).



## Appendix F

# Limiting Behavior of the Turbulent Batch Reactor Model

The turbulent batch reactor model, Eq. (5.20), is equivalent to other approaches to modeling turbulent combustion under certain limiting conditions. These limiting cases are discussed below.

### F.1 Burke-Schumann Solution

When the reactants are initially completely mixed ( $\zeta_0 = 0$ ) and the chemical kinetics are infinitely fast for a single step reaction, Fuel + Air  $\rightarrow$  Products, the present model reduces to the Burke-Schumann solution (see, e.g., [50]), where the cell mean mixture fraction is given by

$$\tilde{Z} = \tilde{Y}_F + \left( \frac{1}{1+s} \right) \tilde{Y}_P. \quad (\text{F.1})$$

### F.2 Basic EDC

When the reactants are initially unmixed ( $\zeta_0 = 1$ ) and the kinetics are infinitely fast, our model reduces to the Eddy Dissipation Concept (EDC) as described in [44]. First, write (5.20) for Fuel (F) with  $\zeta = 1$ :

$$\dot{m}_F''' = \rho \frac{1}{\tau_{\text{mix}}} (\hat{Y}_F - \tilde{Y}_F^0). \quad (\text{F.2})$$

For infinitely fast kinetics, the Fuel composition in the mixed zone is

$$\hat{Y}_F = \begin{cases} 0 & \text{if } \tilde{Y}_F^0 \leq \tilde{Y}_A^0/s \quad (\text{excess Air}), \\ \tilde{Y}_F^0 - \tilde{Y}_A^0/s & \text{if } \tilde{Y}_F^0 > \tilde{Y}_A^0/s \quad (\text{excess Fuel}). \end{cases} \quad (\text{F.3})$$

Using (F.3) in (5.30) we recover the basic EDC model:

$$\dot{m}_F''' = -\rho \frac{\min(\tilde{Y}_F^0, \tilde{Y}_A^0/s)}{\tau_{\text{mix}}}. \quad (\text{F.4})$$

### F.3 Extended EDC

When the unmixed fraction is held constant ( $\zeta = \zeta_0$ ), our formulation may be cast as an extended EDC model with the mixed reactor zone treated as a perfectly stirred reactor (PSR). Previous authors [47, 48, 145] have

referred to the mixed reactor zone as the “fine structure region.” In the notation of [48], the extended EDC reaction rate model, comparable to (5.20) in this paper, is given by

$$\rho \frac{d\tilde{Y}_\alpha}{dt} = \rho \frac{\gamma_\lambda^2 \chi}{\tau^*} (Y_\alpha^0 - Y_\alpha^*) , \quad (\text{F.5})$$

where  $\gamma_\lambda$  is the “ratio between the mass of the fine structure and the total mass of the subgrid structure” and  $\chi$  is the probability of fine structure burning [48]; in practice, both quantities are between 0 and 1. The reactor residence time is denoted  $\tau^*$ , clearly comparable to our mixing time scale, and the fine structure mass fraction is  $Y_\alpha^*$ , clearly comparable to our mixed zone mass fraction  $\hat{Y}_\alpha$ . Panjwani et al. [48] refer to  $Y_\alpha^0$  as the mass fraction of the “surrounding state,” thus equivalent to our initial cell mean  $\tilde{Y}_\alpha^0$ .

## Appendix G

# Scalar Boundedness Correction

Second-order central differencing of the advection term in the scalar transport equation leads to dispersion errors (spurious wiggles) and these errors, if left untreated, can lead to scalar fields which are physically not realizable, e.g., negative densities. To prevent this, FDS employs a boundedness correction to the scalar fields after the explicit transport step. The correction, which we describe below, acts locally and effectively adds the minimum amount of diffusion necessary to prevent boundedness violations. This correction does not make the scalar transport scheme total variation diminishing (TVD); it only serves to correct for boundedness. Similar schemes are employed by others (e.g., [146]).

By default, FDS employs a TVD transport scheme (Superbee [19] for VLES and CHARM [20] for DNS and LES). These TVD schemes are applied during the transport step and each can be shown to be TVD in one dimension under certain CFL constraints. However, except for Godunov's scheme, the TVD proofs do not extend to three dimensions [21]. Still, these schemes do a much better job than pure central differencing at mitigating dispersion error. Note that even though TVD schemes are applied, FDS still checks for boundedness in case any small violations are not prevented by the flux limiter.

### A simple case

For simplicity we start by considering a minimum boundedness violation for density in 1-D. That is, somewhere we have  $\rho < \rho_{\min}$ . Let  $\rho_i^*$  denote the resulting density from the explicit transport step for cell  $i$  with volume  $V_i$ . Our goal is to find a correction  $\delta\rho_i$  which:

- (a) satisfies boundedness,  $\rho_i = \rho_i^* + \delta\rho_i \geq \rho_{\min}$  for all  $i$
- (b) conserves mass,  $\sum_i \delta\rho_i V_i = 0$
- (c) minimizes data variation,  $\sum_i |\delta\rho_i|$  (i.e., we change the field as little as possible)

The basic idea is to apply a linear smoothing operator,  $\mathcal{L}$ , to the density field in regions where boundedness violations have occurred. So, the correction may be viewed as an explicit diffusion step applied to the uncorrected field with diffusion coefficient  $c$ :

$$\rho = \rho^* + c \mathcal{L}(\rho^*) \quad (\text{G.1})$$

To make matters simple, let us envision for the moment that the density in cell  $i$  is negative, but that the densities in cells  $i - 1$  and  $i + 1$  are both safely in bounds (this actually is what happens most of the time with dispersion error). We therefore want a correction that takes mass away from  $i - 1$  and  $i + 1$  and moves it to  $i$  to make up the deficit. We know that for cell  $i$  the minimum change in mass and therefore the minimum

correction that will satisfy boundedness is  $\delta\rho_i = \rho_{\min} - \rho_i^*$ . The operator  $\mathcal{L}$  takes the form of the standard discrete Laplacian. The correction for cell  $i$  is simply

$$\begin{aligned}\rho_i &= \rho_i^* + \delta\rho_i \\ &= \rho_i^* + \rho_{\min} - \rho_i^* \\ &= \rho_i^* + c_i(\rho_{i-1}^* - 2\rho_i^* + \rho_{i+1}^*)\end{aligned}\tag{G.2}$$

Comparing the second and third lines, we find that the diffusion coefficient is given by

$$c_i = \frac{\rho_{\min} - \rho_i^*}{\rho_{i-1}^* - 2\rho_i^* + \rho_{i+1}^*}\tag{G.3}$$

Based on the third line of (G.2), the correction for cell  $i$  may be thought of as the sum of the two mass fluxes from its neighboring cells. The change in mass of cell  $i$  is  $\delta m_i = \delta\rho_i V_i$  and is balanced by changes in mass for cells  $i-1$  and  $i+1$ :

$$\begin{aligned}\delta m_{i-1} &= -c_i(\rho_{i-1}^* - \rho_i^*)V_i \\ \delta m_{i+1} &= -c_i(\rho_{i+1}^* - \rho_i^*)V_i\end{aligned}$$

In this case the sum of the mass corrections is zero, as desired:

$$\begin{aligned}\sum_{j=i-1}^{i+1} \delta m_j &= \delta\rho_{i-1}V_{i-1} + \delta\rho_iV_i + \delta\rho_{i+1}V_{i+1} \\ &= -c_i(\rho_{i-1}^* - \rho_i^*)V_i + c_i(\rho_{i-1}^* - 2\rho_i^* + \rho_{i+1}^*)V_i - c_i(\rho_{i+1}^* - \rho_i^*)V_i \\ &= 0\end{aligned}$$

### Realistic cases

The discussion above was to provide a simple case for understanding the basic idea behind the correction method. In a realistic case we must account for multi-dimensional aspects of the problem and for the possibility that neighboring cells may both be out of bounds. Consider a grid cell whose density is outside the specified range. Denote this cell with a “ $c$ ” for center. Its volume is  $V_c$  and density is  $\rho_c^*$ , obtained from the transport scheme. Let the subscript “ $n$ ” denote any of the six neighboring cells (in other words, only include cells which share a face with cell  $c$ ). We want to correct any boundedness violations for the cell  $c$  by shifting mass to or from its neighboring cells  $n$ :

$$\rho_c = \rho_c^* + \delta\rho_c \quad ; \quad \rho_n = \rho_n^* + \delta\rho_n\tag{G.4}$$

We first define the total amount of mass we wish to shift:

$$m_c = |\rho_c^* - \rho_{\text{cut}}|V_c\tag{G.5}$$

where  $\rho_{\text{cut}}$  is the appropriate upper or lower bound of the density. The amount of mass each neighboring cell can accommodate without falling outside the range is:

$$m_n = \left| \min \left[ \rho_{\max}, \max[\rho_{\min}, \rho_n^*] \right] - \rho_{\text{cut}} \right| V_n\tag{G.6}$$

The correction terms that guarantee mass conservation ( $V_c \delta\rho_c = -\sum V_n \delta\rho_n$ ) are:

$$\delta\rho_c = \pm \min [m_c, \sum m_n] / V_c \quad ; \quad \delta\rho_n = \mp \min \left[ \frac{m_c}{\sum m_n}, 1 \right] m_n / V_n\tag{G.7}$$

Next, to correct species mass fractions that are out of bounds, we follow the exact same procedure.

$$Z_c = Z_c^* + \delta Z_c \quad ; \quad Z_n = Z_n^* + \delta Z_n \quad (\text{G.8})$$

We define the amount of species mass we wish to shift:

$$m_c = |Z_c^* - Z_{\text{cut}}| \rho_c V_c \quad (\text{G.9})$$

where  $Z_{\text{cut}}$  is either 0 or 1. The amount of species mass each neighboring cell can accommodate without falling outside the range is:

$$m_n = \left| \min \left[ 1, \max[0, Z_n^*] \right] - Z_{\text{cut}} \right| \rho_n V_n \quad (\text{G.10})$$

The correction terms that guarantee mass conservation ( $V_c \rho_c \delta Z_c = - \sum V_n \rho_n \delta Z_n$ ) are:

$$\delta Z_c = \pm \min [m_c, \sum m_n] / (\rho_c V_c) \quad ; \quad \delta Z_n = \mp \min \left[ \frac{m_c}{\sum m_n}, 1 \right] m_n / (\rho_n V_n) \quad (\text{G.11})$$



## Appendix H

# Lilly's Analysis for Turbulent Flows

The constants for the Smagorinsky and Deardorff turbulence models are not wholly empirical. They may be derived by assuming an inertial range Kolmogorov spectrum and “production equals dissipation”. This analysis for Smagorinsky is attributed to D. K. Lilly [30], and may also be found in Sagaut [147]. For Deardorff, this analysis is the solution to Exercise 13.45 in Pope [15].

### Smagorinsky

The assumption that the mean production of subgrid-scale kinetic energy ( $\mathcal{P}_{sgs}$ ) equals the mean dissipation rate ( $\varepsilon$ ) implies (here I am being cavalier with my mean quantities)

$$\mathcal{P}_{sgs} = \varepsilon \quad (\text{H.1})$$

Using (4.15) at high Reynolds number ( $\mu_t \gg \mu$ ) for an incompressible flow ( $\nabla \cdot \mathbf{u} = 0$ ), (H.1) leads to

$$2\nu_t \tilde{S}_{ij} \tilde{S}_{ij} = 2\nu S_{ij} S_{ij} \quad (\text{H.2})$$

The components of the strain tensor  $S_{ij}$  are greater in magnitude than the filtered strain components  $\tilde{S}_{ij}$ , and so the turbulent viscosity,  $\nu_t$ , must be larger than the molecular viscosity  $\nu$ , as would be expected.

We assume the filter width,  $\Delta$ , is in the inertial subrange of a high Reynolds number flow. There the Kolmogorov kinetic energy spectrum may be written as

$$E(k) = C_K \varepsilon^{2/3} k^{-5/3} \quad (\text{H.3})$$

displaying the famous “minus five-thirds” scaling. The Kolmogorov constant is usually take to be  $C_K \approx 1.5$ .

Using the filtered dissipation spectrum (see Pope [15]), the Kolmogorov spectrum (H.3), and assuming a spectral cutoff filter we may write

$$2\nu \tilde{S}_{ij} \tilde{S}_{ij} = 2\nu \int_0^{\frac{\pi}{\Delta}} k^2 E(k) dk \quad (\text{H.4})$$

$$= 2\nu C_K \varepsilon^{2/3} \int_0^{\frac{\pi}{\Delta}} k^{1/3} dk \quad (\text{H.5})$$

Dividing by  $\nu$  gives

$$2\tilde{S}_{ij} \tilde{S}_{ij} = 2C_K \varepsilon^{2/3} \frac{3}{4} \left[ k^{4/3} \right]_0^{\frac{\pi}{\Delta}} \quad (\text{H.6})$$

$$|\tilde{S}|^2 = C_K \varepsilon^{2/3} \frac{3}{2} \left( \frac{\pi}{\Delta} \right)^{4/3} \quad (\text{H.7})$$

where  $|\tilde{S}| \equiv (2\tilde{S}_{ij}\tilde{S}_{ij})^{1/2}$ . This allows us to write the dissipation rate in terms of the filtered strain magnitude:

$$\varepsilon = \left( \frac{2}{3C_K} \right)^{3/2} \left( \frac{\Delta}{\pi} \right)^2 |\tilde{S}|^3 \quad (\text{H.8})$$

Now, we invoke production equals dissipation. Below, the left-hand side is the production of SGS kinetic energy. The right-hand side is the dissipation rate in terms of the filtered strain from (H.8):

$$2v_t \tilde{S}_{ij} \tilde{S}_{ij} = \varepsilon \quad (\text{H.9})$$

$$v_t |\tilde{S}|^2 = \left( \frac{2}{3C_K} \right)^{3/2} \left( \frac{\Delta}{\pi} \right)^2 |\tilde{S}|^3 \quad (\text{H.10})$$

Inserting the constant coefficient Smagorinsky model for the turbulent viscosity,  $v_t = (C_s \Delta)^2 |\tilde{S}|$ , yields

$$(C_s \Delta)^2 |\tilde{S}|^3 = \left( \frac{2}{3C_K} \right)^{3/2} \left( \frac{\Delta}{\pi} \right)^2 |\tilde{S}|^3 \quad (\text{H.11})$$

Finally, using  $C_K = 1.5$  and solving for  $C_s$  gives

$$C_s = \left[ \frac{2}{(3)(1.5)} \right]^{3/4} \frac{1}{\pi} = 0.17 \quad (\text{H.12})$$

## Deardorff

The turbulent viscosity for the Deardorff form of the isotropic eddy viscosity is  $v_t = C_v \Delta k_{sgs}^{1/2}$ . In this section, we derive the value of the constant using Lilly's analysis. To start, we require a relationship between the dissipation rate and the subgrid kinetic energy  $k_{sgs}$ . This relationship does not depend on how  $k_{sgs}$  is computed—it depends on the shape of the energy spectrum and on where the filter width cutoff falls in relation to that spectrum. Here we assume a Kolmogorov spectrum at high Re with the filter width falling inside the inertial subrange. Then the subgrid kinetic energy is well approximated by integrating the spectrum from the cutoff wave number to infinity:

$$k_{sgs} = \int_{\frac{\pi}{\Delta}}^{\infty} E(k) dk \quad (\text{H.13})$$

$$= C_K \varepsilon^{2/3} \int_{\frac{\pi}{\Delta}}^{\infty} k^{-5/3} dk \quad (\text{H.14})$$

$$= \frac{3}{2} C_K \varepsilon^{2/3} \left( \frac{\Delta}{\pi} \right)^{2/3} \quad (\text{H.15})$$

which rearranges to

$$\varepsilon = \left( \frac{2}{3C_K} \right)^{3/2} \frac{\pi}{\Delta} k_{sgs}^{3/2} \quad (\text{H.16})$$

Equating the relationships for the dissipation rates from Smagorinsky (H.8) and Deardorff (H.16) gives

$$\left( \frac{\Delta}{\pi} \right)^2 |\tilde{S}|^3 = \frac{\pi}{\Delta} k_{sgs}^{3/2} \quad (\text{H.17})$$

$$|\tilde{S}|^2 = \left( \frac{\pi}{\Delta} \right)^2 k_{sgs} \quad (\text{H.18})$$

Now, we again invoke production equals dissipation using Deardorff as the eddy viscosity model and (H.16) for the dissipation rate:

$$v_t |\tilde{S}|^2 = \epsilon \quad (\text{H.19})$$

$$C_v \Delta k_{sgs}^{1/2} \left( \frac{\pi}{\Delta} \right)^2 k_{sgs} = \left( \frac{2}{3C_K} \right)^{3/2} \frac{\pi}{\Delta} k_{sgs}^{3/2} \quad (\text{H.20})$$

Using  $C_K = 1.5$  and solving for  $C_v$  yields

$$C_v = \left[ \frac{2}{(3)(1.5)} \right]^{3/2} \frac{1}{\pi} = 0.094 \quad (\text{H.21})$$

FDS uses a value of  $C_v = 0.1$  for the Deardorff model constant.



## Appendix I

# Fluid-Particle Momentum Transfer

The trajectory of a Lagrangian particle is governed by the momentum conservation equation:

$$m_p \frac{d\mathbf{u}_p}{dt} = -\frac{1}{2} \rho C_d A_{p,c} (\mathbf{u}_p - \mathbf{u}) \|\mathbf{u}_p - \mathbf{u}\| + m_p \mathbf{g} \quad (I.1)$$

where  $m_p$  is the particle mass,  $\mathbf{u}_p(t)$  the particle velocity,  $A_{p,c}$  the particle cross-sectional area,  $C_d$  the drag coefficient,  $\rho$  the gas density,  $\mathbf{u}$  the gas velocity in the vicinity of the particle, and  $\mathbf{g}$  the gravity vector. There is no analytical solution to this equation, but its linearized form:

$$\frac{d\mathbf{u}_p}{dt} = -\beta (\mathbf{u}_p - \mathbf{u}) + \mathbf{g} \quad \beta = \frac{1}{2m_p} \rho C_d A_{p,c} \|\mathbf{u}_p(0) - \mathbf{u}\| \quad (I.2)$$

has the solution:

$$\mathbf{u}_p(t) = \mathbf{u} + \left( \mathbf{u}_p(0) - \mathbf{u} - \frac{\mathbf{g}}{\beta} \right) e^{-\beta t} + \frac{\mathbf{g}}{\beta} \quad (I.3)$$

assuming that the gas velocity  $\mathbf{u}$  is unchanging over the short duration for which this solution is valid.

In FDS, the particle position is advanced over the course of a gas-phase time step,  $\delta t$ , by a series of sub-time steps,  $\delta t_p$ , that are determined so as to ensure that the particle does not traverse the width of a grid cell in one sub-time step:

$$\delta t_p = \frac{\delta t}{\lceil 0.9 \text{CFL} \rceil} \quad ; \quad \text{CFL} = \delta t \max \left( \frac{|u_p^n|}{\delta x}, \frac{|v_p^n|}{\delta y}, \frac{|w_p^n|}{\delta z} \right) \quad (I.4)$$

Note that the *ceiling* function,  $\lceil \text{CFL} \rceil$ , denotes the least integer greater than the CFL. For a given time step, denoted by  $n$ , the particle position is advanced according to:

$$\mathbf{x}_p^{n+1} = \mathbf{x}_p^n + \frac{\delta t_p}{2} (\mathbf{u}_p^{n+1} + \mathbf{u}_p^n) \quad (I.5)$$

where  $\mathbf{u}_p^{n+1}$  is given by Eq. (I.3). The transfer of momentum from the particles to the gas is represented by the force term in the momentum equation which is computed by summing over all particles in each grid cell of volume  $V$ :

$$\mathbf{f}_b = \frac{1}{V} \sum_i \left[ m_{p,i} \left( \mathbf{g} - \frac{\mathbf{u}_{p,i}^{n+1} - \mathbf{u}_{p,i}^n}{\delta t} \right) + \dot{m}_{p,i} (\mathbf{u}_p^{n+1} - \mathbf{u}^n) \right] \quad (I.6)$$



## Appendix J

# Simplifications of the Radiation Transport Equation

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The Radiation Transport Equation (RTE) for particles is given by Eq. (6.42) and repeated here:

$$\mathbf{s} \cdot \nabla I_\lambda(\mathbf{x}, \mathbf{s}) = -[\kappa_p(\mathbf{x}, \lambda) + \sigma_p(\mathbf{x}, \lambda)] I_\lambda(\mathbf{x}, \mathbf{s}) + \kappa_p(\mathbf{x}, \lambda) I_{b,p}(\mathbf{x}, \lambda) + \frac{\sigma_p(\mathbf{x}, \lambda)}{4\pi} \int_{4\pi} \Phi(\mathbf{s}, \mathbf{s}') I_\lambda(\mathbf{x}, \mathbf{s}') d\mathbf{s}' \quad (\text{J.1})$$

An accurate computation of the in-scattering integral on the right hand side of Eq. (J.1) would be extremely time consuming and require a prohibitive amount of memory because the individual intensities in each location would have to be stored. The in-scattering integral can be approximated by dividing the total  $4\pi$  solid angle into a “forward angle,”  $\delta\Omega^l$ , and an “ambient angle,”  $\delta\Omega^* = 4\pi - \delta\Omega^l$ . For compatibility with the FVM solver,  $\delta\Omega^l$  is set equal to the control angle given by the angular discretization. However, it is assumed to be symmetric around the center of the control angle. Within  $\delta\Omega^l$  the intensity is  $I_\lambda(\mathbf{x}, \mathbf{s})$  and elsewhere it is approximated as

$$U^*(\mathbf{x}, \lambda) = \frac{U(\mathbf{x}, \lambda) - \delta\Omega^l I_\lambda(\mathbf{x}, \mathbf{s})}{\delta\Omega^*} \quad (\text{J.2})$$

where  $U(\mathbf{x})$  is the total intensity integrated over the unit sphere. The in-scattering integral in Eq. (J.1) can now be approximated as

$$\begin{aligned} \frac{\sigma_p(\mathbf{x}, \lambda)}{4\pi} \int_{4\pi} \Phi(\mathbf{s}, \mathbf{s}') I_\lambda(\mathbf{x}, \mathbf{s}') d\mathbf{s}' &\approx \sigma_p(\mathbf{x}, \lambda) \left( \chi_f I_\lambda(\mathbf{x}, \mathbf{s}) + \frac{1}{\delta\Omega^*} (1 - \chi_f) \int_{\delta\Omega^*} I_\lambda(\mathbf{x}, \mathbf{s}') d\mathbf{s}' \right) \\ &= \sigma_p(\mathbf{x}, \lambda) \left( \chi_f I_\lambda(\mathbf{x}, \mathbf{s}) + (1 - \chi_f) U^*(\mathbf{x}, \lambda) \right) \end{aligned} \quad (\text{J.3})$$

where  $\chi_f = \chi_f(r, \lambda) = \chi_f(\mathbf{x}, \lambda)$  is a fraction of the total intensity originally within the solid angle  $\delta\Omega^l$  that is scattered into the same angle,  $\delta\Omega^l$ . The calculation of  $\chi_f$  is discussed in section 6.3.3. Using the definition of  $U^*$  in Eq. (J.2), the RTE is now:

$$\mathbf{s} \cdot \nabla I_\lambda(\mathbf{x}, \mathbf{s}) = -\kappa_p(\mathbf{x}, \lambda) I_\lambda(\mathbf{x}, \mathbf{s}) + \kappa_p(\mathbf{x}, \lambda) I_{b,p}(\mathbf{x}, \lambda) + \sigma_p(\mathbf{x}, \lambda) (1 - \chi_f) \left( \frac{U(\mathbf{x}, \lambda) - \delta\Omega^l I_\lambda(\mathbf{x}, \mathbf{s})}{\delta\Omega^*} - I_\lambda(\mathbf{x}, \mathbf{s}) \right) \quad (\text{J.4})$$

An effective scattering coefficient is next defined

$$\bar{\sigma}_p(\mathbf{x}, \lambda) = \frac{4\pi}{4\pi - \delta\Omega^l} (1 - \chi_f(\mathbf{x}, \lambda)) \sigma_p(\mathbf{x}, \lambda) \quad (\text{J.5})$$

Using Eq. (J.5) and  $\delta\Omega^* = 4\pi - \delta\Omega^l$ , the scattering coefficient can be written as:

$$\sigma_p(\mathbf{x}, \lambda) = \frac{\bar{\sigma}_p(\mathbf{x}, \lambda)}{4\pi} \frac{\delta\Omega^*}{(1 - \chi_f)} \quad (\text{J.6})$$

Substituting this in the previous form of the RTE yields:

$$\mathbf{s} \cdot \nabla I_\lambda(\mathbf{x}, \mathbf{s}) = -\kappa_p(\mathbf{x}, \lambda) I_\lambda(\mathbf{x}, \mathbf{s}) + \kappa_p(\mathbf{x}, \lambda) I_{b,p}(\mathbf{x}, \lambda) + \frac{\bar{\sigma}_p(\mathbf{x}, \lambda)}{4\pi} \left( U(\mathbf{x}, \lambda) - (\delta\Omega^l + \delta\Omega^*) I_\lambda(\mathbf{x}, \mathbf{s}) \right) \quad (\text{J.7})$$

Using  $\delta\Omega^l + \delta\Omega^* = 4\pi$ , the RTE simplifies to:

$$\mathbf{s} \cdot \nabla I_\lambda(\mathbf{x}, \mathbf{s}) = -[\kappa_p(\mathbf{x}, \lambda) + \bar{\sigma}_p(\mathbf{x}, \lambda)] I_\lambda(\mathbf{x}, \mathbf{s}) + \kappa_p(\mathbf{x}, \lambda) I_{b,p}(\mathbf{x}, \lambda) + \frac{\bar{\sigma}_p(\mathbf{x}, \lambda)}{4\pi} U(\mathbf{x}, \lambda) \quad (\text{J.8})$$

## Appendix K

# Absorption Coefficients of Liquid Fuels

The burning rate of liquid pool fires depends in part on the convective and radiative heat feedback from the flames to the fuel surface. For large pool fires, radiation heat transfer dominates. Studies have been conducted to determine the spectra of emitted radiation [148] as well as to characterize radiation absorption by gases within the flame [149]. Depending on the fuel, thermal radiation can be absorbed at the surface or in depth. For fuels such as wood, most of the incident radiation is absorbed within a thin layer near the surface. For semi-transparent materials such as plastics or liquid fuels, thermal radiation may penetrate deeper. The in-depth radiation absorption by semi-transparent fuels has been studied for PMMA [150], polymer films [151] and liquid pool fires [148]. Most of the research related to the in-depth radiation absorption in liquids considers the boil-over of liquid pool fires on water [152]. The effect of in-depth radiation absorption on evaporation of fuel droplets has also received some attention [153]. Most liquids are highly selective absorbers, absorbing intensively in some wavelength regions while being transparent in others. This results in radiation transport models that are both computationally expensive and for which experimental data is scarce. In this appendix, we attempt to characterize the absorption of radiation by liquid fuels using effective absorption coefficients similar to those used in Refs. [154] and [155].

Where data on absorption coefficients of liquids exists in the open literature, such as the Coblenz Society data found on the NIST Chemistry WebBook [156], it usually only contains data for wavelengths from approximately  $2.5\text{ }\mu\text{m}$  upwards. A large part of the total energy in the emission spectrum of flames may easily be contained in wavelengths shorter than  $2.5\text{ }\mu\text{m}$ . However, absorption spectra that begin from  $1\text{ }\mu\text{m}$  exists for a few liquids, including toluene ([157], [158]), methanol [159], benzene [160] and water [161]. Furthermore, Ref. [148] includes spectrally resolved transmission spectra of ethanol, heptane, JP-8, and an ethanol-toluene blend. Complex refractive index spectra for a few diesel fuels were reported in Ref. [153]. Different diesel fuels have slightly different absorption spectra, due to differing additives. However, the data reported in Ref. [153] can perhaps be used to obtain an order of magnitude estimate for the absorption coefficient of diesel fuel.

Often we are not interested in resolving the spectra of the transmitted radiation; rather, we are interested in modeling the total transmitted radiation. In these cases, it is convenient to write the radiation transport equations in terms of mean absorption coefficients. This is done to avoid the time-consuming integrations over all wavelengths. For this reason, a number of mean absorption coefficients have been introduced, such as the Rosseland-mean absorption coefficient and the Planck-mean absorption coefficient. These correspond to the optically-thin approximation and the Rosseland diffusion approximation of radiation transport. The absorption coefficients of liquids are highly wavelength dependent and are even transparent in some areas. In this case, the Planck-mean absorption coefficients are too large by several orders of magnitude. It is preferable to determine an effective absorption coefficient that attempts to replicate the absorption of radiation over a certain path length over which the majority of the radiation is absorbed.

Table K.1 lists effective absorption coefficients for a few selected liquids. It contains two types of absorption coefficients. One type is determined by assuming the incoming radiation is blackbody radiation at a temperature of 1450 K. The other type is based on actual flame radiation spectra. If the wavenumber range is listed, then a blackbody temperature 1450 K is assumed in calculating the transmission. If the wavenumber range is not listed, then transmission data from Ref. [148] is used. The path length is 3 mm in all cases.

The assumption of blackbody radiation is adequate for sooty flames in which radiation from soot dominates the flame radiation spectra. However, for fuels with low sooting flames, the incoming radiation spectrum differs considerably from the blackbody spectrum. This explains the large difference in listed absorption coefficients for ethanol and methanol. The ethanol absorption coefficient is based on actual emission spectra of an ethanol flame, whereas the methanol absorption coefficient is calculated based on the blackbody spectrum. The correct value for methanol is likely to be closer to that of ethanol.

Table K.1: Effective absorption coefficients for selected liquids.

Liquid	Wavenumber Range (1/cm)	Effective Absorption Coefficient, $\kappa$
JP-8 [148]	-	301.4
Ethanol-Toluene blend [148]	-	680.1
Ethanol [148]	-	1534.3
Heptane [148]	-	187.5
Toluene [158]	436-6500	160.8
Methanol[159]	2-8000	52
Water [161]	1-15000	1578
Benzene [160]	11.5-6200	123

## Appendix L

# Solving the Heat Conduction Equation

By default, FDS solves the 1-D heat conduction equation in the direction normal to a solid surface. There is also an optional 3-D solver that makes use of the 1-D solver in an alternating direction implicit (ADI) scheme.

### L.1 One-Dimensional Heat Conduction in Solids

The 1-D heat conduction equation in Cartesian coordinates is:

$$\rho_s c_s \frac{\partial T_s}{\partial t} = \frac{\partial}{\partial x} \left( k_s \frac{\partial T_s}{\partial x} \right) + \dot{q}_s''' \quad (\text{L.1})$$

In cylindrical and spherical coordinates, the equation becomes:

$$\rho_s c_s \frac{\partial T_s}{\partial t} = \frac{1}{r^I} \frac{\partial}{\partial r} \left( r^I k_s \frac{\partial T_s}{\partial r} \right) + \dot{q}_s''' \quad (\text{L.2})$$

where  $I$  is 1 for cylindrical and 2 for spherical coordinates. Since for  $I = 0$ , the Cartesian formulation is recovered, from this point on we will consider the general form of the equation (L.2). The indexing system used for the discretization of the equations is shown in Fig. L.1. The discretization scheme is the same regardless of whether the coordinate system is Cartesian, cylindrical, or spherical. The size of the cell nearest the boundary is set by default to be

$$x_1 - x_0 = \sqrt{\frac{\delta t_0 k_s}{\rho_s c_s}} \quad (\text{L.3})$$

where  $\delta t_0$  is a somewhat arbitrary time scale set to 1. In other words, the Fourier number based on the cell size is assumed of order unity. The user has flexibility in changing both the initial cell size and the degree of stretching.

The temperature at the center of each solid cell,  $T_{s,i}$ , is updated in time using a Crank-Nicolson scheme:

$$\begin{aligned} (\rho_s c_s)_i \frac{T_{s,i}^{n+1} - T_{s,i}^n}{\delta t} &= \frac{1}{2 r_{c,i}^I \delta r_i} \left( r_i^I k_{s,i+\frac{1}{2}} \frac{T_{s,i+1}^n - T_{s,i}^n}{\delta r_{i+\frac{1}{2}}} - r_{i-1}^I k_{s,i-\frac{1}{2}} \frac{T_{s,i}^n - T_{s,i-1}^n}{\delta r_{i-\frac{1}{2}}} \right) \\ &\quad + \frac{1}{2 r_{c,i}^I \delta r_i} \left( r_i^I k_{s,i+\frac{1}{2}} \frac{T_{s,i+1}^{n+1} - T_{s,i}^{n+1}}{\delta r_{i+\frac{1}{2}}} - r_{i-1}^I k_{s,i-\frac{1}{2}} \frac{T_{s,i}^{n+1} - T_{s,i-1}^{n+1}}{\delta r_{i-\frac{1}{2}}} \right) + \dot{q}_{s,i}''' \end{aligned} \quad (\text{L.4})$$

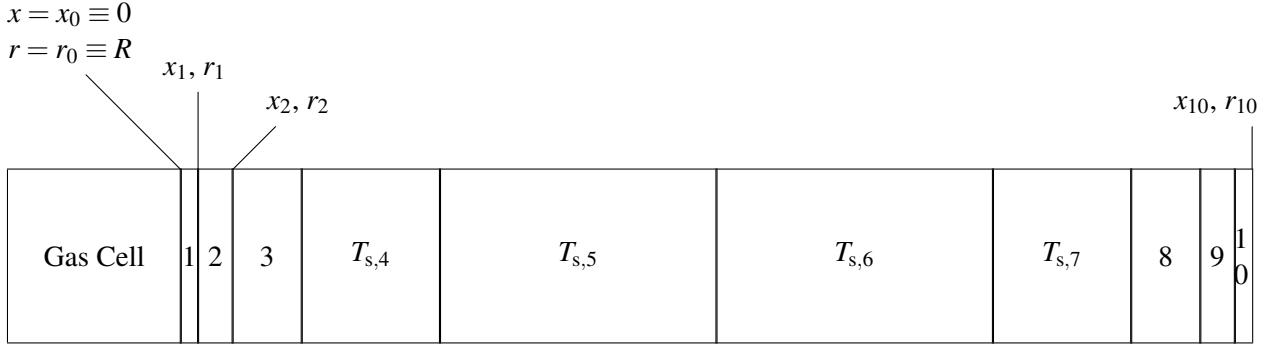


Figure L.1: Layout of solid phase cells. In this example, each solid phase cell is stretched by a factor of 2 until the halfway point of the solid is reached, after which the cells shrink by a factor of 2. The interface between gas and solid phase is at  $x_0 = R - r_0$ , where  $R$  is the radius of cylinder or sphere. The back side boundary condition is applied at  $x_{10} = R - r_{10}$ . “Virtual” cells 0 and  $N + 1 = 11$  are located to the left and right of the solid, and are the same size as cells 1 and  $N = 10$ .

$k_{s,i+\frac{1}{2}}$  is the thermal conductivity at the border of the cells  $i$  and  $i + 1$ ,  $\delta r_i$  is the width of cell  $i$ , and  $\delta r_{i+\frac{1}{2}}$  is the distance from the center of cell  $i$  to the center of cell  $i + 1$ . The radial coordinate,  $r_{c,i}$ , denotes the cell center:

$$r_{c,i}^I = \begin{cases} (r_i^2 - r_{i-1}^2)/(2\delta r_i) & I = 1 \quad (\text{cylindrical}) \\ \sqrt{(r_i^3 - r_{i-1}^3)/(3\delta r_i)} & I = 2 \quad (\text{spherical}) \end{cases} \quad (\text{L.5})$$

The temperatures at the front and back surface of a Cartesian slab (or center of a cylinder or sphere) are determined from the boundary conditions. In any of the coordinate systems, the boundary condition at the front surface, using the Crank-Nicolson scheme, is

$$\frac{-k_{s,1}}{2} \left( \frac{T_{s,1}^{n+1} - T_{s,0}^{n+1}}{\delta r_{\frac{1}{2}}} + \frac{T_{s,1}^n - T_{s,0}^n}{\delta r_{\frac{1}{2}}} \right) = (\dot{q}_c'' + \dot{q}_r'')^{n+\frac{1}{2}} \quad (\text{L.6})$$

Note that  $T_{s,0}$  does not represent the gas temperature, but rather it is used to establish the temperature gradient at the surface. Note also that the thermal conductivity,  $k_{s,1}$ , is that of the first solid cell. The convective heat flux at the mid-point of the time step is

$$(\dot{q}_c'')^{n+\frac{1}{2}} = \frac{1}{2} \left( h^n \left( T_g^n - T_{s,\frac{1}{2}}^n \right) + h^{n+1} \left( T_g^{n+1} - T_{s,\frac{1}{2}}^{n+1} \right) \right) \quad (\text{L.7})$$

where  $T_g^n$  and  $T_g^{n+1}$  are the gas temperatures in the first grid cell abutting the surface at the start and end of the time step,  $T_{s,\frac{1}{2}}$  is the surface temperature, defined as the average of  $T_{s,0}$  and  $T_{s,1}$ , and  $h^n$  and  $h^{n+1}$  are the convective heat transfer coefficients at the start and end of the time step. The radiative heat flux can be estimated at the mid-point of the time step:

$$(\dot{q}_r'')^{n+\frac{1}{2}} = \dot{q}_{r,in}'' - \varepsilon \sigma \left( T_{s,\frac{1}{2}}^{n+\frac{1}{2}} \right)^4 \approx \dot{q}_{r,in}'' - \varepsilon \sigma \left( T_{s,\frac{1}{2}}^n \right)^4 - 2\varepsilon \sigma \left( T_{s,\frac{1}{2}}^n \right)^3 \left( T_{s,\frac{1}{2}}^{n+1} - T_{s,\frac{1}{2}}^n \right) \quad (\text{L.8})$$

Note the use of Taylor series expansion:

$$(T^{n+1/2})^4 \approx (T^n)^4 + 2(T^n)^3(T^{n+1} - T^n) \quad (\text{L.9})$$

Now the front boundary condition is

$$-k_{s,1} \frac{T_{s,1}^{n+1} - T_{s,0}^{n+1} + T_{s,1}^n - T_{s,0}^n}{2\delta r_{\frac{1}{2}}} \approx \frac{1}{2} \left( h^n \left( T_g^n - T_{s,\frac{1}{2}}^n \right) + h^{n+1} \left( T_g^{n+1} - T_{s,\frac{1}{2}}^{n+1} \right) \right)$$

$$+ \dot{q}_{r,in}'' - 2\varepsilon\sigma \left(T_{s,\frac{1}{2}}^n\right)^3 T_{s,\frac{1}{2}}^{n+1} + \varepsilon\sigma \left(T_{s,\frac{1}{2}}^n\right)^4 \quad (\text{L.10})$$

The wall surface temperature is defined:

$$T_{s,\frac{1}{2}} = \frac{T_{s,1} + T_{s,0}}{2} \quad (\text{L.11})$$

and the temperature gradient at the start of the time step is:

$$k_{s,1} \frac{T_{s,1}^n - T_{s,0}^n}{2\delta r_{\frac{1}{2}}} = -\frac{h^n}{2} \left(T_g^n - T_{s,\frac{1}{2}}^n\right) - \frac{1}{2} \left(\dot{q}_{r,in}'' - \varepsilon\sigma \left(T_{s,\frac{1}{2}}^n\right)^4\right) \quad (\text{L.12})$$

and therefore the boundary condition becomes

$$-k_{s,1} \frac{T_{s,1}^{n+1} - T_{s,0}^{n+1}}{2\delta r_{\frac{1}{2}}} + \left(\frac{h^{n+1}}{2} + 2\varepsilon\sigma \left(T_{s,\frac{1}{2}}^n\right)^3\right) \frac{T_{s,1}^{n+1} + T_{s,0}^{n+1}}{2} \approx \frac{1}{2} (h^{n+1} T_g^{n+1} + \dot{q}_{r,in}'') + \frac{3}{2} \varepsilon\sigma \left(T_{s,\frac{1}{2}}^n\right)^4 \quad (\text{L.13})$$

Rearranging terms, the temperature at node 0 becomes:

$$T_{s,0}^{n+1} = \underbrace{\frac{k_{s,1}}{\delta r_{\frac{1}{2}}} - \left(\frac{h^{n+1}}{2} + 2\varepsilon\sigma \left(T_{s,\frac{1}{2}}^n\right)^3\right)}_{\text{RFACF2}} T_{s,1}^{n+1} + \underbrace{\frac{h^{n+1} T_g^{n+1} + \dot{q}_{r,in}'' + 3\varepsilon\sigma \left(T_{s,\frac{1}{2}}^n\right)^4}{\frac{k_{s,1}}{\delta r_{\frac{1}{2}}} + \left(\frac{h^{n+1}}{2} + 2\varepsilon\sigma \left(T_{s,\frac{1}{2}}^n\right)^3\right)}}_{\text{QDXKF}} \quad (\text{L.14})$$

Note that `RFACF2` and similar names are used in the actual source code to represent these quantities. In case of non-insulated backing in Cartesian geometry, the temperature of virtual node  $N+1$  is calculated the same way. For a Cartesian geometry with an insulated backing or for cylindrical and spherical geometries,  $\varepsilon$ ,  $\dot{q}_{r,in}''$ , and  $h$  are set to 0.

After re-arranging the terms, Eq. (L.4) becomes (using the nomenclature of the source code) for each wall cell  $i$ :

$$B_i T_{i-1}^{n+1} + D_i T_i^{n+1} + A_i T_{i+1}^{n+1} = C_i \quad i = 1, \dots, N \quad (\text{L.15})$$

where

$$\begin{aligned} A_i &= -\frac{k_{i+\frac{1}{2}} \delta t}{2(\rho_s c_s)_i} \frac{1}{r_{c,i}^I \delta r_i} \frac{r_i^I}{\delta r_{i+\frac{1}{2}}} \\ B_i &= -\frac{k_{i-\frac{1}{2}} \delta t}{2(\rho_s c_s)_i} \frac{1}{r_{c,i}^I \delta r_i} \frac{r_{i-1}^I}{\delta r_{i-\frac{1}{2}}} \\ C_i &= T_{s,i}^n - A_i (T_{i+1}^n - T_i^n) + B_i (T_i^n - T_{i-1}^n) + \delta t \dot{q}_{s,i}''' / (\rho_s c_s)_i \\ D_i &= 1 - A_i - B_i \end{aligned} \quad (\text{L.16})$$

To solve Eq. (L.15), a tri-diagonal linear solver is used:

$$C_1 = C_1 - B_1 \cdot \text{QDXKF}$$

$$C_N = C_N - A_N \cdot \text{QDXKB}$$

$$D_1 = D_1 + B_1 \cdot \text{RFACF2}$$

$$D_N = D_N + A_N \cdot \text{RFACB2}$$

$$D_i = D_i - \frac{B_i A_{i-1}}{D_{i-1}} \quad i = 2, \dots, N-1$$

$$C_i = C_i - \frac{B_i C_{i-1}}{D_{i-1}} \quad i = 2, \dots, N-1 \quad (\text{L.17})$$

$$C_N = \frac{C_N}{D_N}$$

$$T_{s,N}^{n+1} = C_N$$

$$T_{s,i}^{n+1} = \frac{C_i - A_i C_{i+1}}{D_i} \quad i = N-1, \dots, 1$$

Typically, the time step,  $\delta t$ , used by the solid phase solver is some multiple of the time step used by the gas phase solver. By default, the multiple is 2, but the user can select a different value if need be. However, there may be times during the simulation where the solid phase time step must be smaller than the gas phase time step due to potentially large, short-lived excursions of the internal heat source term,  $\dot{q}_s'''$ , which is not included in the implicit Crank-Nicolson scheme. It is simply evaluated at the start of the time step. To remedy this problem, after the heat source term has been evaluated, the maximum temperature change is estimated using an explicit, first order update:

$$\Delta T_s^* \equiv \max_{i=1,N} \left| \frac{\delta t}{(\rho_s c_s)_i} \left[ \frac{1}{r_{c,i}^I \delta r_i} \left( r_i^I k_{i+\frac{1}{2}} \frac{T_{s,i+1}^n - T_{s,i}^n}{\delta r_{i+\frac{1}{2}}} - r_{i-1}^I k_{s,i+\frac{1}{2}} \frac{T_{s,i}^n - T_{s,i-1}^n}{\delta r_{i-\frac{1}{2}}} \right) + \dot{q}_{s,i}''' \right] \right| \quad (\text{L.18})$$

If  $\Delta T_s^*$  exceeds a user-specified threshold,  $\Delta T_{\max}$ , the time step is reduced below the gas phase time step,  $\delta t_{\text{gas}}$ :

$$\delta t = \delta t_{\text{gas}} 2^{-[\ln \eta / \ln 2]} \quad ; \quad \eta = \Delta T_s^* / \Delta T_{\max} \quad (\text{L.19})$$

Note that  $[x]$  denotes the least integer greater than  $x$ . To prevent an endless spiral of decreasing solid phase time steps, there is a user-specified limit of the number of subdivisions of the gas phase time step. The default is  $2^2 = 4$ .

## Dirichlet Temperature Boundaries

For the purposes of verification, it is sometimes useful to hold the surface temperature of a solid fixed. In this case, the boundary condition is of Dirichlet type and Eq. (L.6) defines a constant wall temperature,  $T_w$ ,

$$T_{s,\frac{1}{2}} = \frac{T_{s,1} + T_{s,0}}{2} = T_w \quad (\text{L.20})$$

The ghost cell value at  $n + 1$  may then be written in terms of the wall temperature and the temperature of the first cell,

$$T_{s,0}^{n+1} = -T_{s,1}^{n+1} + 2T_w \quad (\text{L.21})$$

Comparing Eq. (L.21) with Eq. (L.14), it is apparent that

$$\text{RFACF2} = -1 \quad (\text{L.22})$$

$$\text{QDXKF} = 2T_w \quad (\text{L.23})$$

## L.2 Optional 3-D Solid Phase Heat Transfer

The solid phase heat conduction equation in three directions is given by:

$$\rho_s c_s \frac{\partial T_s}{\partial t} = \frac{\partial}{\partial x} \left( k_s \frac{\partial T_s}{\partial x} \right) + \frac{\partial}{\partial y} \left( k_s \frac{\partial T_s}{\partial y} \right) + \frac{\partial}{\partial z} \left( k_s \frac{\partial T_s}{\partial z} \right) + \dot{q}_s''' \quad (\text{L.24})$$

The method of solution is to split this equation using the following alternating direction implicit (ADI) scheme:

$$\frac{\rho_s c_s}{3} \frac{\partial T_s}{\partial t} = \frac{\partial}{\partial x} \left( k_s \frac{\partial T_s}{\partial x} \right) + \frac{\dot{q}_s'''}{3} \quad (\text{L.25})$$

$$\frac{\rho_s c_s}{3} \frac{\partial T_s}{\partial t} = \frac{\partial}{\partial y} \left( k_s \frac{\partial T_s}{\partial y} \right) + \frac{\dot{q}_s'''}{3} \quad (\text{L.26})$$

$$\frac{\rho_s c_s}{3} \frac{\partial T_s}{\partial t} = \frac{\partial}{\partial z} \left( k_s \frac{\partial T_s}{\partial z} \right) + \frac{\dot{q}_s'''}{3} \quad (\text{L.27})$$

The solution of the equation is updated over the course of three gas phase time steps of size  $\delta t$ . For each coordinate direction, the temperature field is finely gridded near the surface and coarsely gridded in the interior, typically. Thus, the temperature field,  $T_s(x, y, z, t)$ , is discretized differently in each direction. For example, for each surface cell normal to the  $x$  direction, the 1-D temperature array in depth is  $T_{s,x,i}$ , where  $1 \leq i \leq N_x$ , and  $N_x$  is the number of non-uniformly spaced internal cells spanning the entire width of the solid in the  $x$  direction. This is exactly the same approach taken above for 1-D heat conduction in solids with “exposed” back surfaces. Likewise,  $T_s(x, y, z, t)$  is discretized  $T_{s,y,j}$  and  $T_{s,z,k}$  in the  $y$  and  $z$  directions, respectively.

In each of the three steps, the 1-D heat conduction equation is solved in one of the three coordinate directions using the Crank-Nicolson (C-N) scheme described above, except now the implicit update spans three gas phase time steps,  $3\delta t$ . The temperature fields spanning the other two coordinate directions are updated explicitly. The change in temperature due to the implicit C-N update is denoted  $\Delta T_{s,x,i}$ , with similar expressions for the  $y$  and  $z$  updates. The explicit update taken during the two steps for which  $T_{s,x,i}$  is not updated implicitly transfers enthalpy in the  $y$  and  $z$  directions:

$$\rho_{s,i} c_{s,i} T_{s,x,i}^{n+1} = \rho_{s,i} c_{s,i} T_{s,x,i}^n + \frac{1}{2} \left[ \sum_j w_j \rho_{s,j} c_{s,j} \Delta T_{s,y,j} + \sum_k w_k \rho_{s,k} c_{s,k} \Delta T_{s,z,k} \right] \quad (\text{L.28})$$

The coefficients,  $w_j$  and  $w_k$ , are the fractions of cell  $i$  of the  $x$  direction discretization overlapped by cells  $j$  and  $k$  of the  $y$  and  $z$  discretizations, respectively.

This type of splitting scheme is discussed by Toro [21], Chapter 16, “Methods for Multi-Dimensional PDEs.”



## Appendix M

# Development of an Implicit Solution for Droplet Evaporation

In this appendix, we consider evaporation of a single drop in a computational cell where the drop may be in the gas or attached to a solid surface. Hence, our goal is to develop a solution for (a) the cell water vapor mass fraction  $Y_{\alpha,g}$ , (b) the cell gas phase temperature  $T_g$ , (c) the temperature of the droplet  $T_p$ , and (d) if present an estimate of the first node of a solid surface  $T_w$ .

The ODEs to be integrated are, respectively, the droplet mass, the change in cell water vapor mass fraction, the change in the droplet temperature, the change in the cell gas phase temperature, and the wall temperature [101]:

$$\frac{dm_p}{dt} = -A_{p,s} h_m \rho_f (Y_{\alpha,\ell} - Y_{\alpha,g}) \quad (\text{M.1})$$

$$\rho_g V \frac{dY_{\alpha,g}}{dt} = - (1 - Y_{\alpha,g}) \frac{dm_p}{dt} \quad (\text{M.2})$$

$$\frac{dT_p}{dt} = \frac{1}{m_p c_p} \left[ \dot{q}_r + A_{p,s} h_g (T_g - T_p) + A_{p,s} h_w (T_w - T_p) + \frac{dm_p}{dt} h_v \right] \quad (\text{M.3})$$

$$\frac{dT_g}{dt} = \frac{1}{m_g c_g} \left[ A_{p,s} h_g (T_p - T_g) - \frac{dm_p}{dt} (h_{\alpha,p} - h_{\alpha,g}) \right] \quad (\text{M.4})$$

$$\frac{dT_w}{dt} = - \frac{A_{p,s} h_w}{m_w c_w} (T_w - T_p) \quad (\text{M.5})$$

Below we will consider the time integration for a single substep in the evaporation routine, which updates the local computational cell from time  $t^n$  to  $t^n + \Delta t^n$ . Usually,  $\Delta t^n = \Delta t_{\text{LES}}$ . However, it is possible that subtimesteps may be required to prevent spurious mass and temperature overshoots. In what follows, unless otherwise specified RHS quantities are evaluated at the beginning of the substep,  $t^n$  (hence the designation “semi-implicit” method). This limits the stability range of the method (since the method is not fully implicit it is not unconditionally stable), but it greatly simplifies the solution procedure.

To begin the development, we expanded Eq. (M.2) with the mass loss rate (to be determined) evaluated at the midpoint of the time interval:

$$Y_{\alpha,g}^{n+1} = Y_{\alpha,g}^n - \frac{\Delta t^n}{\rho_g V} (1 - Y_{\alpha,g}) \left( \frac{dm_p}{dt} \right)^{n+\frac{1}{2}} \quad (\text{M.6})$$

Substituting Eq. M.6 into a Crank-Nicolson expansion of Eq. (M.1) we have

$$\begin{aligned}
\left( \frac{dm_p}{dt} \right)^{n+\frac{1}{2}} &= -\frac{A_{p,s}h_m\rho_f}{2} \left( Y_{\alpha,\ell}^{n+1} + Y_{\alpha,\ell}^n - Y_{\alpha,g}^{n+1} - Y_{\alpha,g}^n \right) \\
&= -\frac{A_{p,s}h_m\rho_f}{2} \left( Y_{\alpha,\ell}^{n+1} + Y_{\alpha,\ell}^n + \frac{\Delta t^n}{\rho_g V} (1 - Y_{\alpha,g}) \left( \frac{dm_p}{dt} \right) - 2Y_{\alpha,g}^n \right) \\
&= -\frac{A_{p,s}h_m\rho_f}{2} \left[ \frac{Y_{\alpha,\ell}^{n+1} + Y_{\alpha,\ell}^n - 2Y_{\alpha,g}^n}{1 + \frac{\Delta t^n A_{p,s}h_m (1 - Y_{\alpha,g})}{2\rho_g V}} \right]
\end{aligned} \tag{M.7}$$

Next, we find an approximation for the droplet surface equilibrium mass fraction  $Y_{\alpha,\ell}^{n+1}$  via the expansion

$$Y_{\alpha,\ell}^{n+1} = Y_{\alpha,\ell}^n + \left( \frac{dY_{\alpha,\ell}}{dT} \right) (T_p^{n+1} - T_p^n) \tag{M.8}$$

The derivative is determined from a chain rule expansion:

$$\left( \frac{dY_{\alpha,\ell}}{dT} \right) = \left( \frac{dY_{\alpha,\ell}}{dX_{\alpha,\ell}} \right) \left( \frac{dX_{\alpha,\ell}}{dT} \right) \tag{M.9}$$

The water vapor mole fraction on the surface of the droplet is determined from the Clausius-Clapeyron equation:

$$X_{\alpha,\ell} = \exp \left[ \frac{h_v W_\alpha}{R} \left( \frac{1}{T_b} - \frac{1}{T_p} \right) \right] \tag{M.10}$$

For a binary mixture in air, the relationship between the vapor mass and mole fractions is given by

$$Y_{\alpha,\ell} = \frac{X_{\alpha,\ell}}{X_{\alpha,\ell}(1 - W_a/W_\alpha) + W_a/W_\alpha} \tag{M.11}$$

Differentiating Eqs. (M.10) and (M.11) and substituting into (M.9) yields

$$\left( \frac{dY_{\alpha,\ell}}{dT} \right) = \frac{W_a/W_\alpha}{(X_{\alpha,\ell}(1 - W_a/W_\alpha) + W_a/W_\alpha)^2} \frac{h_v W_\alpha}{RT_p^2} \exp \left[ \frac{h_v W_\alpha}{R} \left( \frac{1}{T_b} - \frac{1}{T_p} \right) \right] \tag{M.12}$$

Using (M.12) in (M.8), subsequently (M.8) in (M.7), and then (M.7) in (M.3) and (M.4), leaves three equations with three unknowns,  $T_p^{n+1}$ ,  $T_g^{n+1}$ , and  $T_w^{n+1}$ , to be solved simultaneously. As previously noted, unless otherwise indicated, all RHS quantities are evaluated at the beginning of the substep,  $t^n$ .

$$\begin{aligned}
T_p^{n+1} &= T_p^n + \frac{\Delta t^n}{m_p c_p} \left[ \dot{q}_r + \frac{A_{p,s}h_g}{2} (T_g^{n+1} + T_g^n - T_p^{n+1} - T_p^n) \right. \\
&\quad \left. + \frac{A_{p,s}h_w}{2} (T_w^{n+1} + T_w^n - T_p^{n+1} - T_p^n) - \frac{A_{p,s}h_m\rho_f}{2} \left[ \frac{Y_{\alpha,\ell}^{n+1} + Y_{\alpha,\ell}^n - 2Y_{\alpha,g}^n}{1 + \frac{\Delta t^n A_{p,s}h_m (1 - Y_{\alpha,g})}{2\rho_g V}} \right] h_v \right]
\end{aligned} \tag{M.13}$$

$$\begin{aligned}
T_g^{n+1} &= T_g^n + \frac{\Delta t^n}{m_g c_g} \left[ \frac{A_{p,s}h_g}{2} (T_p^{n+1} + T_p^n - T_g^{n+1} - T_g^n) \right. \\
&\quad \left. + \frac{A_{p,s}h_m\rho_f}{2} \left[ \frac{Y_{\alpha,\ell}^{n+1} + Y_{\alpha,\ell}^n - 2Y_{\alpha,g}^n}{1 + \frac{\Delta t^n A_{p,s}h_m (1 - Y_{\alpha,g})}{2\rho_g V}} \right] (h_{\alpha,p} - h_{\alpha,g}) \right]
\end{aligned} \tag{M.14}$$

$$T_w^{n+1} = T_w^n - \frac{\Delta t^n A_{p,s} h_w}{2m_w c_w} [T_w^{n+1} + T_w^n - T_p^{n+1} - T_p^n] \quad (\text{M.15})$$

Finally, the updated cell water vapor mass fraction,  $Y_{\alpha,g}^{n+1}$ , may be found using (M.8) in (M.7), and then (M.7) in (M.6).

$$Y_{\alpha,g}^{n+1} = Y_{\alpha,g}^n + \frac{\Delta t^n A_{p,s} h_m \rho_f}{\rho V} \left[ \frac{2Y_{\alpha,\ell}^n + \left( \frac{dY_{\alpha,\ell}}{dT} \right) (T_p^{n+1} - T_p^n) - 2Y_{\alpha,g}^n}{1 + \frac{\Delta t^n A_g h_m (1 - Y_{\alpha,g})}{2\rho_g V}} \right] \quad (\text{M.16})$$



## Appendix N

# A Special Preconditioning Scheme for Solving the Poisson Equation in Tunnels

Consider the Poisson equation in a tunnel geometry where multiple meshes abut end to end in the  $x$  direction. Start with the conventional equation for the pressure term,  $H(\mathbf{x})$ :

$$\nabla^2 H = -\frac{\partial(\nabla \cdot \mathbf{u})}{\partial t} - \nabla \cdot \mathbf{F} \equiv f(\mathbf{x}) \quad (\text{N.1})$$

with boundary conditions

$$\frac{\partial H}{\partial n} = b_{n,m} \quad \text{or} \quad H = b_{n,m} \quad ; \quad n \in (x, y, z) \quad m \in (1, 2) \quad (\text{N.2})$$

at each of the six boundary faces of the rectangular domain. Now assume that  $H$  can be decomposed:

$$H(\mathbf{x}) = \bar{H}(x) + H'(\mathbf{x}) \quad (\text{N.3})$$

where  $x$  is the lengthwise spatial coordinate of the tunnel. Solve the following equation for  $\bar{H}$ :

$$\frac{d^2 \bar{H}}{dx^2} = \bar{f}(x) \quad ; \quad \bar{f}(x) = \frac{1}{\bar{Y}\bar{Z}} \int_0^{\bar{Y}} \int_0^{\bar{Z}} f(\mathbf{x}) dz dy \quad (\text{N.4})$$

with boundary conditions at the ends of the tunnel:

$$\frac{d\bar{H}}{dx} = \bar{b}_{x,m} \quad \text{or} \quad \bar{H} = \bar{b}_{x,m} \quad ; \quad \bar{b}_{x,m} = \frac{1}{\bar{Y}\bar{Z}} \int_0^{\bar{Y}} \int_0^{\bar{Z}} b_{x,m}(y, z) dz dy \quad (\text{N.5})$$

Now solve in each mesh using the 3-D Poisson solver:

$$\nabla^2 H' = f(\mathbf{x}) - \bar{f}(x) \quad (\text{N.6})$$

with boundary conditions at the left and right face of each mesh, including the ends of the tunnel:

$$\frac{\partial H'}{\partial x} = b_{x,m}(y, z) - \bar{b}_{x,m} \quad \text{or} \quad H' = b_{x,m}(y, z) - \bar{b}_{x,m} \quad (\text{N.7})$$

The averaging term,  $\bar{H}$ , handles the significant fluctuations in pressure that are caused by the fluctuations in the fire's heat release rate.  $H'$  is much easier to solve for because it does not include the large pressure fluctuations.

To implement this in FDS, the terms  $\bar{f}$  and  $\bar{b}_{x,m}$  are gathered to the root MPI process which solves the tri-diagonal linear system resulting from discretizing Eq. (N.4). The resulting solution is then broadcast to the other MPI processes and added to the solution of Eq. (N.6) which is computed on each individual mesh independently. The decomposition and reconstruction of  $H$  is all done within the pressure solving step, reducing the memory requirements to just the 1-D array of  $\bar{H}$ .



# Appendix O

## Energy and Mass Conservation for Solid Phase Reactions

A solid material in FDS can undergo one or more reactions (see Sec. 7.2) that produce new solid materials, gases, or Lagrangian particles (for example, melting or brand formation). These reactions should conserve both mass and energy.

### O.1 Mass Conservation

Mass conservation for reaction  $\beta$  for solid material  $\alpha$  is given by:

$$\sum_{\alpha'=1}^{N_m} v_{\alpha',\alpha\beta} = 0 \quad (\text{O.1})$$

where  $N_m$  is the number of all types of substances in FDS (solid, gas, or particle) that the reaction produces and  $v_{\alpha',\alpha\beta}$  is the yield of  $\alpha'$  in the reaction where  $\alpha'$  could be a new material, a gas, or a type of Lagrangian particle. In FDS input processing, the yield of the original material,  $v_{\alpha,\alpha\beta}$ , is taken as -1, and the prior equation written in terms of the yields as specified in an input file becomes:

$$\sum_{\alpha'=1, \alpha' \neq \alpha}^{N_m} v_{\alpha',\alpha\beta} = 1 \quad (\text{O.2})$$

If the left hand side were to sum to larger than 1, this would mean mass is being created. This is not permitted and will result in an error message. If the left hand side sums to less than 1, this would mean mass is being destroyed. This, however, only leads to a warning message. Loosing mass is allowed to represent phenomena where mass is lost from the solid but the user does not wish to track what happens to that mass in the domain.

### O.2 Energy Conservation

For energy conservation, the enthalpy prior to the reaction must equal the enthalpy after the reaction plus the heat of reaction,  $H_{r,\alpha\beta}$ . This, in fact, defines the heat of reaction at a temperature as the difference in enthalpy between the products and reactants:

$$\sum_{\alpha'=1, v_{\alpha',\alpha\beta} < 0}^{N_m} -v_{\alpha',\alpha\beta} h_{\alpha'}(T) = \sum_{\alpha'=1, v_{\alpha',\alpha\beta} > 0}^{N_m} v_{\alpha',\alpha\beta} h_{\alpha'}(T) + H_{r,\alpha\beta}(T) \quad (\text{O.3})$$

Ensuring energy conservation requires that two things be performed as part of handling solid phase reactions. First, the temperature dependence of the heat of reaction must be determined. Second, in reactions combining gasses and solids (including particles), enthalpy differences due to temperature differences must be correctly assessed and apportioned to the gas and solid phase. A challenge for the first item is that for many real world materials, detailed enthalpy data is not known. Enthalpy is known for many pure solid substances (such as NaCl or SiO<sub>2</sub>), but not for other types of solids like wood or many polymers (plastics and foams). If enthalpies and specific heats are known for all the components of a reaction, then a temperature dependent heat of reaction can be easily generated as the temperature dependent enthalpy difference between products and reactants.

It is noted that the temperature dependence of the heat of reaction is generally a minor effect. For many pyrolysis reactions of interest to fire, the heat of reaction is hundreds of kJ/kg and differences in the specific heat of products and reactants are generally less than 1 kJ/(kg K). In many cases a specific reaction occurs over a narrow temperature range of a few tens of K. Splitting that temperature range around the temperature of the peak reaction rate means that over the temperature range where a reaction takes place that the heat of reaction is often being adjusted by under 10 %.

### O.2.1 Solid Phase Heat of Reaction

For the first step, reference enthalpies for all the components in a reaction must be determined. FDS has a database for many common gases that includes predefined reference enthalpies. In a typical simulation any gases not in the database are likely fuels participating in combustion reactions where the reference enthalpy can be determined from the heat of combustion and the reference enthalpies of the combustion products. Therefore, gas reference enthalpies are treated as known quantities for the solid phase. A similar rationale applies to liquid particles produced by a solid phase reaction. Definition of a liquid particle includes specifying the gas it evaporates to and the heat of vaporization which means a reference enthalpy for the liquid can be determined. Reference enthalpies for solid materials are generally not known except for some pure substances. Even if available, the published values when placed in Eq. O.3 may not exactly result in the specified heat of reaction. Reference enthalpies for any solid material participating as a reactant or product in a solid phase reaction must be determined. For solid materials not participating in reactions the assumption that the reference enthalpy is 0 kJ/kg at 0 K suffices as there is no exchange of mass (and its associated enthalpy) with the gas phase. Solid phase products for a solid phase reaction include both solid materials that remain part of the reacting surface and solid materials that leave the surface in the form of particles (i.e., a firebrand). Finding these enthalpies is done by solving a system of linear equations where there is one equation for every material reaction and one unknown for the reference enthalpy of every solid material participating in a material reaction as reactant or product. To build the matrix, Eq. O.3 is applied to every material reaction. The enthalpy of material  $\alpha$ ,  $h_\alpha$ , is given by [26]:

$$h_\alpha(T) = h_{\alpha,\text{ref}} + \int_{T_{\text{ref}}}^{T_R} c_{p,\alpha}(T) dT \quad (\text{O.4})$$

In this equation  $h_{\alpha,\text{ref}}$  is the enthalpy at the temperature  $T_{\text{ref}}$ . This is the value being solved for. In many tabulations of solid enthalpies the value of  $T_{\text{ref}}$  is 298.15 K; however, since it can be determined at any reference temperature, for convenience a temperature of 0 K is picked. The specific heat of a material,  $c_{p,\alpha}(T)$ , is defined in the input file. Evaluating the integral requires picking the upper bound temperature,  $T_R$ . This is taken as the reference temperature for the reaction. This may be user defined as part of the input. If not, it can be determined from the reaction  $A_{\alpha\beta}$  and  $E_{\alpha\beta}$  by performing a TGA simulation for material  $\alpha$ , reaction  $\beta$ . The value for  $T_R$  is set to the temperature of the peak reaction rate. With  $T_R$  defined, the various  $v_{\alpha',\alpha\beta} h_{\alpha,\text{ref}}$  terms can be collected on the left hand side, and the heats of reaction and the various  $v_{\alpha',\alpha\beta} \int_{T_{\text{ref}}}^{T_R} c_{p,\alpha}(T) dT$  terms can be collected on the right hand side. The result is a system of linear equations

$A\bar{x} = \bar{b}$  where the unknowns,  $\bar{x}$ , are the values of  $h_{\alpha,\text{ref}}$ ; however, it is not guaranteed that there are an equal number of reactions and unknown material reference enthalpies. Solving the system of equations depends on the relative numbers of equations and unknowns as follows:

**Reactions = unknowns** Solution can be found by a simple solution of the linear system:  $\bar{x} = A^{-1}\bar{b}$

**Reactions < unknowns** In this case there are an infinite number of solutions; however, any single solution will suffice. The solution that gives the minimum vector magnitude of  $\bar{x}$  is  $\bar{x} = A^T(AA^T)^{-1}\bar{b}$ .

**Reactions > unknowns** In this case the system is overdetermined, and there may not be a solution vector  $\bar{x}$  that will perfectly conserve energy. A least-squares solution is given by  $\bar{x} = (A^TA)^{-1}A^T\bar{b}$ .

The first two cases are typically expected. The first case is where all materials decomposes into one or more gas species and there is no material residue. The second case is where one or more materials produce one or more residues including the possibility that the residue has a reaction. For example, wood pyrolyzing into a fuel gas plus char where the char reacts with oxygen to produce and ash remnant and CO<sub>2</sub>. The third case would be where multiple solid materials are reactants in the same reaction. This case, while possible, is not typical for pyrolysis reactions.

If there is a reaction where mass is destroyed (the LHS of Eq. (O.2) is less than 1), this approach must be modified. An assumption must be made about the destroyed mass so that the reaction  $v$  values balance. In this case, if a reaction loses mass, FDS assumes that the lost mass has the same temperature dependent specific heat as the original material. FDS then adds a virtual material for that product which is included in the solution vector. The virtual material is not tracked outside of determining reference enthalpies.

Once the reference enthalpies are known, the heat of reaction at a specific temperature can be determined by applying Eq. O.3 at that temperature and solving for the heat of reaction.

### O.2.2 Accounting for Solid Cell and Gas Cell Temperature Differences

The second aspect of energy conservation during a solid phase reaction is accounting for the different temperatures at which reactants and products may exist. In FDS the solid phase reaction occurs at the temperature of the solid cell being evaluated. Therefore, if the reaction consumes a gas, that gas must first be brought to the solid temperature. If the gas is hotter or cooler, an energy source or sink term will exist in the solid phase. Reaction products are produced at the solid temperature. For products that are other solid materials, nothing needs to be done as their solid temperature remains the same. For a Lagrangian particle, particles are produced at the solid temperature as well. Again, no further adjustments are needed. Note, that since particles may not be created every time step, FDS keeps track of the particle mass and particle enthalpy created by material reactions. At the next particle injection time, the particle is created at a temperature that conserves the accumulated enthalpy. For a gas product, the reaction produces the gas at the solid temperature. In FDS, the gas is injected as if it is at the current gas cell temperature. Therefore, the opposite adjustment must be made for gases that are produced as it made for gases that are consumed. In the gas phase if the current gas cell temperature is hotter than the solid, an energy sink term must be generated. In both cases the term is the mass flow rate in kg/s times the difference in enthalpy of the produced or consumed material between the gas temperature and the solid temperature (kJ/kg).

$$\dot{q}_{\alpha'}''' = r_{\alpha\beta} v_{\alpha',\alpha\beta} (h_{\alpha'}(T_g) - h_{\alpha'}(T_s)) \quad (\text{O.5})$$

This term is added to the solid cell enthalpy for gases that are consumed ( $v_{\alpha',\alpha\beta} < 0$  when  $\alpha'$  is a gas). This term is subtracted from the gas cell enthalpy for the gases that are produced ( $v_{\alpha',\alpha\beta} > 0$  when  $\alpha'$  is a gas). When a lumped species is specified for a material reaction, FDS will attempt to determine what primitive sub-species is actually involved in the reaction. For example, consider a solid phase combustion

reaction where the oxygen in the air lumped species plus a material is converted to products. Assuming the products species is appropriately defined, FDS can examine the difference in the masses of primitive species to determine that oxygen is consumed from air but the other components (e.g., ambient N<sub>2</sub>, CO<sub>2</sub>, and H<sub>2</sub>O) are simply passed through to the products side. In this case the enthalpy correction term for the solid would only include the O<sub>2</sub> consumption, and the enthalpy correction term for the gas would only include those products in excess of the original air minus its oxygen.

## Appendix P

# The Scaling Pyrolysis (SPyro) Method

This approach preserves the shape of the heat release rate with time (i.e., burning behavior) measured from cone calorimeter experiments but scales the magnitude and duration based on incident heat at the surface. The model is able to scale a single reference curve to different thermal exposures and material thicknesses. An empirical estimation of the flame heat flux is used to calculate the total incident heat flux in a cone calorimeter for use in scaling. Improved agreement can be achieved by incorporating data from multiple cone calorimeter experiments at different thermal exposures and/or material thicknesses. The following subsections describe the technical basis of each of these modules.

### P.1 Fixed Thickness Scaling Model

The fixed thickness scaling based pyrolysis model is presented in [162]. The net heat flux into the material is related to the pyrolysis rate through the effective heat of gasification according to the equation

$$\dot{q}_{\text{net}}'' = \dot{m}_f'' \Delta h_g \quad (\text{P.1})$$

where  $\dot{q}_{\text{net}}''$  is the net heat flux into the surface and  $\Delta h_g$  is the heat of gasification. Similarly, the heat release rate per unit area of a material,  $\dot{Q}''$ , is related to the pyrolysis rate of the material through the effective heat of combustion,  $\Delta h$ , as

$$\dot{Q}'' = \dot{m}_f'' \Delta h \quad (\text{P.2})$$

The ratio of Eq. (P.1) and Eq. (P.2) relate  $\dot{q}_{\text{net}}''$  and  $\dot{Q}''$  as

$$\frac{\dot{Q}''}{\dot{q}_{\text{net}}''} = \frac{\Delta h}{\Delta h_g} \quad (\text{P.3})$$

Assuming  $\Delta h_g$  and  $\Delta h$  are invariant with  $\dot{q}_{\text{net}}''$ , Eq. (P.3) can be related to predict the change in magnitude of  $\dot{Q}''$  due to changes in thermal exposure as

$$\dot{Q}_1'' = \frac{\dot{q}_{\text{net},1}''}{\dot{q}_{\text{net},2}''} \dot{Q}_2'' \quad (\text{P.4})$$

where the subscripts 1 and 2 indicate different thermal exposure levels. The acceleration or deceleration of the burning can be calculated by evaluating the difference in time required to release the same amount of energy at the two  $\dot{Q}$  over a discrete time interval,  $\Delta t$ ,

$$E_{\Delta t}'' = \dot{q}_{\text{net}}'' \frac{\Delta h}{\Delta h_g} \Delta t \quad (\text{P.5})$$

where  $E_{\Delta t}$  is the total energy released over  $\Delta t$ . Relating Eq. (P.5) at two thermal exposure levels yields

$$\Delta t_1 = \frac{\dot{q}_{\text{net},2}''}{\dot{q}_{\text{net},1}''} \Delta t_2 \quad (\text{P.6})$$

Because  $\dot{q}_{\text{net}}''$  is difficult to measure during burning, an approximate reference heat flux,  $\dot{q}_{\text{ref}}''$ , is introduced as

$$\dot{q}_{\text{ref}}'' = \dot{q}_{\text{cone}}''(1 - \Gamma) + \dot{q}_{\text{flame}}'' \quad (\text{P.7})$$

where  $\dot{q}_{\text{cone}}''$  is the set exposure flux to a cold surface in the cone calorimeter testing,  $\Gamma$  is the fraction of that flux absorbed by the flame, and  $\dot{q}_{\text{flame}}''$  is the heat feedback from the flame to the surface. For the input data from a cone test, the values of  $\Gamma$  and  $\dot{q}_{\text{ref}}''$  are not readily obtained, and the empirical approach discussed in Section P.3 is used to estimate them when determining  $\dot{q}_{\text{net},2}''$  in Eq. P.4. For  $\dot{q}_{\text{net},1}''$ , FDS uses the sum of the incident radiation and the net convective heat flux in the calculation at each time step.

## P.2 Adaptation to Multiple Thicknesses

Detailed pyrolysis models are typically based on an Arrhenius kinetics formulation [163]:

$$r = A e^{-E/(RT_s)} \zeta_1 \zeta_2 \dots \zeta_N \quad (\text{P.8})$$

where  $r$  is the reaction rate,  $A$  and  $E$  are the Arrhenius pre-exponential factor and activation energy,  $T_s$  is the solid temperature,  $R$  is the molar gas constant, and  $\zeta_j$  is the impact of the  $j$ -th reactant species concentration on the burning. There are two primary impacts of material thickness on the burning behavior of a material. The largest impact is to the burning duration due to differences in the combustible mass. The second impact is the transition between a thermally thin and thermally thick material. The model for scaling to different material thicknesses takes inspiration from this formulation; however, the model assumes that the two impacts can be represented by a non-dimensional scaling of a measured burning profile.

The non-dimensional temperature profile within a solid material scales with the Fourier, Fo, and Biot, Bi, numbers. Fo is the ratio of the heat conduction rate to the rate of thermal energy storage in a solid, and can be thought of as a dimensionless time [75]. It is defined as

$$\text{Fo} = \frac{\alpha}{\Delta^2} t = \frac{k}{\rho c_p \Delta^2} t \quad (\text{P.9})$$

where  $\alpha$  is the thermal diffusivity,  $\rho$  is the density,  $c_p$  is the specific heat capacity,  $k$  is the thermal conductivity,  $\Delta$  is the thickness, and  $t$  is time. Bi is the ratio of the internal thermal resistance of a solid to the external resistance of the surface boundary layer [75]:

$$\text{Bi} = \frac{h}{k/\Delta} \quad (\text{P.10})$$

where  $h$  is the heat transfer coefficient. The traditional formulation of Bi assumes that the external exposure of the solid is related primarily to surface convection and thus uses the convection heat transfer coefficient,  $h_c$ , for  $h$ . However, this assumption is not valid in the case of burning solid fuels where external radiation (such as the applied surface heat flux in a cone calorimeter experiment) and flame radiation from the combustion of pyrolyzate above the surface drive the heat transfer to the surface. This is accounted for with an alternative Biot number,  $\text{Bi}^*$ , which combines the effect of surface convection and radiation:

$$\text{Bi}^* = \frac{h_c + h_r}{k/\Delta} \quad (\text{P.11})$$

where  $h$  has been split into two components for convection,  $h_c$ , and radiation,  $h_r$ . The radiation heat transfer coefficient is calculated based on the linearized formulation [75]:

$$h_r = \varepsilon \sigma (T_r + T_s) (T_r^2 + T_s^2) \quad (\text{P.12})$$

where  $\varepsilon$  is the surface emissivity,  $\sigma$  is the Stefan-Boltzmann constant,  $T_r$  is the radiative temperature, and  $T_s$  is the surface temperature. The radiative temperature is an effective temperature representative of the energy absorbed at the surface assuming a view factor of 1:

$$T_r = \left[ \frac{\dot{q}_r''}{\varepsilon \sigma} \right]^{\frac{1}{4}} \quad (\text{P.13})$$

where  $\dot{q}_r''$  is the incident radiation to the surface.

As discussed in Section P.1,  $\Delta h_g$  is assumed to be constant in the prior approach. However, a time-varying effective  $\Delta h_g$  can be calculated based on Eq. (P.3) using the time-varying measurement of  $\dot{Q}''$  and estimating the time-varying  $\dot{q}_{\text{flame}}''$ . It is assumed in this model that the variability in burning intensity and duration observed in experiments can be related to the impact of the material thickness and thermal exposure on these non-dimensional groups,  $\Delta h_g = f(\text{Bi}^*, \text{Fo})$ .  $\Delta h_g$  was found to collapse at the non-dimensional time-scale  $\text{FoBi}^*$ . A scaling relationship for  $\text{FoBi}^*$  can be established by equating this non-dimensional time across two exposure levels,

$$\text{FoBi}^* = \frac{h_1 t_1}{\rho_1 c_{p,1} \Delta_1} = \frac{h_2 t_2}{\rho_2 c_{p,2} \Delta_2} \quad (\text{P.14})$$

where the subscripts indicate different exposure levels. Equation (P.14) reveals that non-dimensional time scales with  $\rho$ ,  $c_p$ ,  $\Delta$ , and  $h$ .

A simplified representation for the scaling of non-dimensional time can be obtained by assuming that changes in material properties (i.e.,  $k$ ,  $\rho$ ,  $c_p$ ) are related to non-dimensional time. Under this assumption, Eq. (P.14) reduces to:

$$t_2 = \frac{\Delta_2}{\Delta_1} \frac{h_1}{h_2} t_1 \quad (\text{P.15})$$

In the case of constant thickness (i.e.,  $\Delta_1 = \Delta_2$ ), Eq. (P.15) is consistent with Eq. (P.6) across a discrete time interval. Thus, the ratio of  $h$  scales similarly to the ratio of  $\dot{q}''$ , yielding:

$$\Delta t = \frac{\Delta}{\Delta_{\text{ref}}} \frac{\dot{q}_{\text{ref}}''}{\dot{q}''} \Delta t_{\text{ref}} \quad (\text{P.16})$$

### P.3 Cone Reference Heat Flux

When using Eq. P.7 to obtain  $\dot{q}_{\text{net},2}''$  in Eq. P.4, an approach is needed to determine  $\Gamma$  and  $\dot{q}_{\text{flame}}''$ . In an actual cone test these will depend upon the material being burned as the heat of combustion, burning rate, and soot yield will all influence the flame heat flux and the absorption of the cone radiation. As there is no simple analytical model to predict these, an empirical approach was developed using FDS simulations of a cone calorimeter.

The simulation geometry is shown in Fig. P.1 and includes the radiant cone, the sample holder, and a volume above the cone to capture the flame height contribution to the heat feedback. A set of 30 simulations were run where the heat of combustion (10, 20, 30, 40, and 50 MJ/kg) and soot yield (0, 1, 2, 5, 10, and 20 %) were varied (CO yield was assumed to equal the soot yield). Each simulation consisted of three phases: the sample burning at a prescribed rate with the cone off, the sample burning at a prescribed rate with the cone on, and no burning with the cone on ( $\dot{q}_{\text{cone}}'' = 50 \text{ kW/m}^2$ ). The prescribed burning was varied

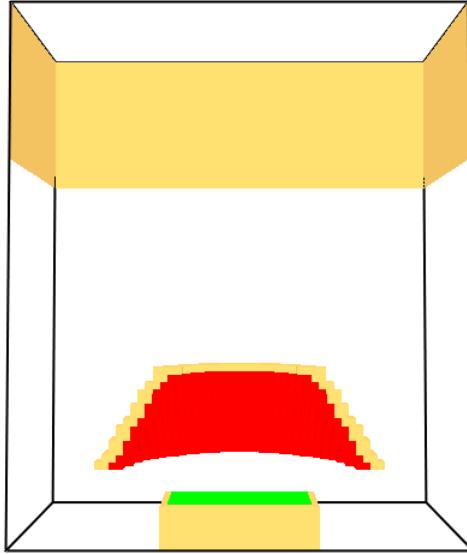


Figure P.1: FDS geometry used for developing the cone reference flux. Red is the cone heater and green is the sample surface. The domain is clipped at the plane  $y=0$  m.

over a range of HRRPUA (100, 200, 400, 800, 1200, 1600, and 2000  $\text{kW}/\text{m}^2$ ). The sample was given a fixed temperature of 300 °C representing a notional ignition temperature. The cone off phase gives  $\dot{q}_{\text{flame}}''$ , the delta between burning with the cone on and burning with the cone off and on gives the cone flux reaching the sample, and the cone with no burning phase gives  $\dot{q}_{\text{cone}}''$ . The difference between the cone flux reaching the sample and  $\dot{q}_{\text{cone}}''$  gives  $\Gamma$ .

A notional fuel molecule was created for each heat of combustion and soot yield that provided an oxygen heat of combustion (EPUMO2) of 13,100 kJ/kg. For a 0 % soot yield, the following was done to obtain fuel chemistry. A heat of combustion of 50 MJ/kg implies a fuel like methane and the formula  $\text{CH}_{3.333}$  provides the desired EPUMO2 at a soot yield of 0 %. A heat of combustion of 40 MJ/kg implies a hydrocarbon like fuel, and the resulting formula is  $\text{CH}_{0.959}$ . For 10, 20 and 30 MJ/kg oxygen was added while keeping a C:H ratio of 1:2, resulting in formulas of  $\text{CH}_2\text{O}_{0.302}$ ,  $\text{CH}_2\text{O}_{0.658}$ , and  $\text{CH}_2\text{O}_{1.322}$ . The H or O values were varied as needed for the different soot yields to keep an EPUMO2 of 13,100 kJ/kg.

If  $\dot{q}_{\text{flame}}''$  for each heat of combustion is normalized by the maximum value for all HRRPUA, the results collapse as shown in the left of Fig P.2. We can then determine a quadratic fit for the maximum  $\dot{q}_{\text{flame}}''$  as a function of heat of combustion as shown in the right of Fig P.2.

Various simple combinations of the heat of combustion, soot yield, and HRRPUA were evaluated to determine a relationship for  $\Gamma$ . The functional form of  $A + B \times \text{HRRPUA}/\Delta h + C \times \text{soot yield}$  gave a fit with the lowest RMS value. To determine the time-dependent  $\dot{q}_{\text{ref}}''$  for a cone test, first, the maximum  $\dot{q}_{\text{flame}}''$  is determined using  $\Delta h$  at the fit shown in Fig P.2, second,  $\dot{q}_{\text{flame}}''$  is adjusted for the current HRRPUA by interpolating the average normalized curve in Fig P.2, third,  $\Gamma$  is determined and used to adjust  $\dot{q}_{\text{cone}}''$ , and finally,  $\dot{q}_{\text{ref}}''$  is determined via Eq. P.7. The resulting fit has a 2 % error compared to the FDS predictions, see Fig. P.3.

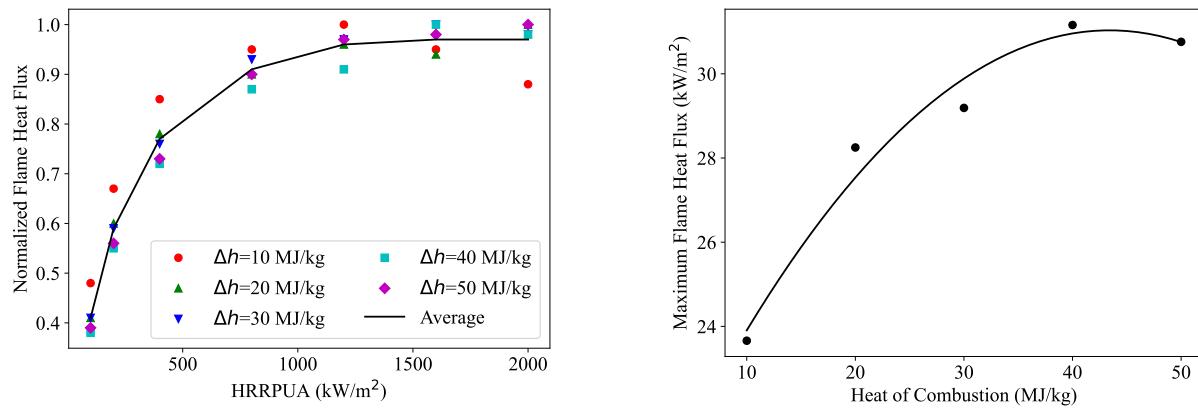


Figure P.2: (Left) Normalized flame heat flux as a function of HRRPUA. (Right) Maximum flame heat flux as a function of  $\Delta h$ .

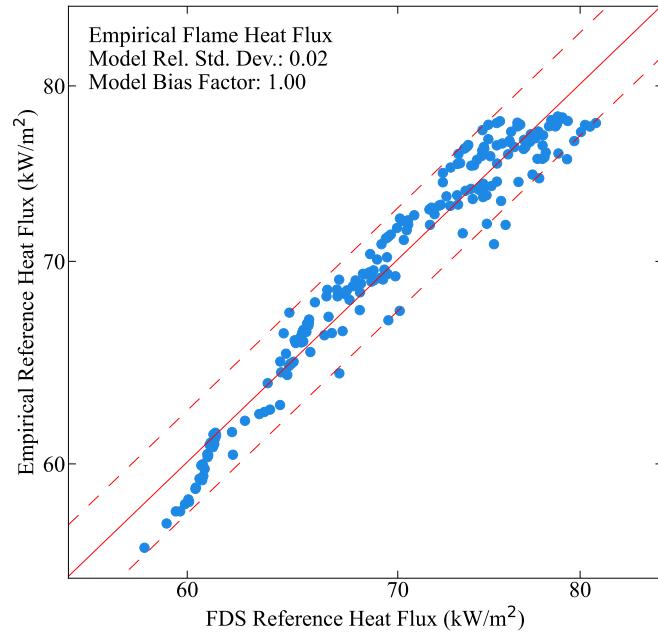


Figure P.3: Scatterplot for predicted vs. empirical reference heat flux.