**Chapter 3**

**Verification**

**3.0 Concept and literature review**

The increasing demand of CFD numerical codes in various scientific and engineering fields calls for methods to establish the accuracy of the numerical scheme. With increases in computational power, CFD practitioners often focus on solving more complex and difficult problems rather than demonstrating the accuracy of their current problems which can lead to a decrease in the quality of their simulations. Previous works on verification and validation in the CFD community [1-6] define verification as demonstrating the mathematical correctness of the numerical simulation. This usually means that if the observed discretization error decreases to zero as the mesh increments decrease to zero, then the equations are “solved correctly”. In other words, code verification is a procedure to demonstrate that the governing equations, as implemented in the code, are solved consistently.

Verification of CFD codes has been the subject of many studies in recent years. Abanto et al [3] demonstrated an approach to test the accuracy of some of the most widespread commercial codes. They presented grid convergence studies on atypical CFD cases using some commercial CFD packages. Their verification test cases include an incompressible laminar Poiseuille flow, a manufactured incompressible laminar boundary layer flow, an incompressible re-circulating flow and an incompressible annular flow. Different types of structured and unstructured meshes were used during the study. They observed non-monotonic grid convergence for all their test cases. Iterative convergence of the discrete equations to machine zero did not guaranty accurate flow field predications which meant that the codes converged to wrong solutions. From their study, they recommended that users perform the verification of commercial CFD codes and be cautious when using the commercial codes on industrial problems.

Kleb and Wood [4] pointed out that the computational simulation community is not routinely publishing independently verifiable tests to accompany new models or algorithms. They mentioned the importance of conducting component-level verification tests before attempting system-level verification and also publishing them when introducing a new component algorithm. They proposed a protocol for the introduction of new methods and physical models that would provide the computational community with a credible history of documented, repeatable verification tests that would enable independent replication.

Roache [1, 5] discussed the verification of codes and calculations along with some definitions and descriptions related to confidence building in computational fluid dynamics. Verification was described as solving the equations right and validation as solving the right equations. Different aspects discussed in the paper include the distinction between code verification and validation, grid convergence and iterative convergence, truncation error and discretization error. Also discussed were verification of calculations, error taxonomies, code verification via systematic grid convergence testing, the Grid Convergence Index (GCI) and sensitivity of grid convergence testing. According to the author, verification does not include all aspects of code quality assurance like the important concerns of version control or archiving of input data. In the book by Roache et al [5], the authors comprehensively discussed code verification, the Method of Manufactured Solutions (MMS) used to obtain exact solutions for code verification purposes, and order of accuracy verification. A more recent study of code verification conducted by Shunn et al [6] demonstrates the MMS verification technique as applied to variable density solvers. In this study, verification was used to investigate the effects of tabulated state-equations and temporal iteration errors on the convergence and accuracy of the code. The two problems constructed were diffusive mixing of species, and convection of density fronts which reflect basic physical phenomena found in combustion problems. The grid refinement studies that were performed confirm that the spatial convergence rate of the solver to be second order when an analytical equation of state (EOS) is used. Convergence of the flow variables to the exact solution were found to be impaired when the EOS was linearly interpolated in space. It was also found that EOS interpolation errors introduce spurious numerical fluctuations in the flow variables, with velocity and pressure being particularly vulnerable. This particular variable density algorithm showed first order temporal evolution of the flow when a single outer density iteration was applied. Temporal errors were generally not dominant when multiple outer density iterations were performed, making it difficult to confirm the temporal accuracy of the method with multiple outer iterations.

**3.1 Verification procedures**

The intent of this chapter is to present a suitable verification framework that can be applied to our own in-house solver, les3d-mp. The procedure can be used to provide a pass-fail acceptance criterion commonly used in the community to establish the validity of CFD solvers [6]. This same procedure is found to be very helpful in detecting coding mistakes (bugs) that are associated with spatial or temporal discretization of the transport equations. The need for verification in this project arises due to the introduction of numerical schemes and models to account for variable density Low-Mach number physics. Numerical schemes and model developments were presented in Chapter 2.

Currently, our verification procedure consists of comparing our computational solution to an exact analytical solution representative of the physics involved in low-mach number problems. Comparing to an exact solution is called MES, or method of exact solutions, and is a powerful verification technique when one can develop an analytical solution for a test case. Comparing to an exact solution brings up the notion of discretization errors which are inherent to any solver that discretizes a set of governing PDEs into a finite dimensional subspace which approximates the continuum solution. The difference between the two is the discretization error. Discretization methods are *consistent* if the error goes to zero as the representative cell size, *h*, decreases to zero (for mesh size *h*, then a consistent method will result in error that is proportional to *hp*). The rate at which the error decreases to zero is called the order of accuracy. A discretization method is said to be second order accurate in space if the discretization error goes to zero as *h2*. According to the community the most rigorous acceptance criterion is verification of order of accuracy, in which one not only seeks to verify that the method is consistent, but also establishes the value of and is then compared to the theoretical order of the discretization method. This is our established procedure.

The following sections present spatial and temporal order of accuracy test cases that have been developed with variable density transport equations in mind. The first part presents cases related to the spatial order of accuracy where we seek to establish the second order accuracy of les3d-mp through a one dimensional isothermal binary mixing case where large density ratios are present. We then seek to show the order of accuracy of the integration scheme options through the solution of a time dependent ODE, . Then, a time-periodic Poiseuille flow is also presented to demonstrate the order of accuracy when . Lastly, two classical test cases corresponding to momentum driven and buoyancy driven laminar boundary layers are presented. Velocity, temperature and near wall property comparisons are made with the analytical Blasius and Ostrach similarity solutions. It is shown that the use of outflow boundary conditions for buoyancy driven boundary layers is not as accurate as for momentum driven boundary layers and the problem is discussed.

**3.2 Spatial accuracy case**

**3.2.1. Isothermal Binary Mixing**

The following verification case tests the ability of the solver to handle flows with large density ratios similar to those found in fires or in combustion systems. An exact solution to the one dimensional mixing of two fluids with different molecular weights is presented. The mixing occurs at constant temperature and pressure conditions. The configuration corresponds to a stratified fluid with a light fluid near a solid wall mixing with a heavy fluid in the ambient. Note that the configuration has zero gravity and is one-dimensional. The computational domain is assumed to be very large in width and depth consistent with using periodic conditions in streamwise and spanwise directions. The restriction of no-slip boundary flow is imposed at the wall while also specifying symmetry conditions for total enthalpy and mixture fraction at the same location. The configuration is presented,

*No-slip velocity*

*Symmetry scalar conditions*

*density decreases*

Figure 1. 1D binary mixing layer. Initial mixing layer thickness is shown by

**3.2.1.1 Governing Equations**

An exact solution for the binary mixing problem is found for the mass density field which directly couples the mixture fraction and velocity. The solution has the characteristic of being a transient mass density diffusion equation satisfying, . The mass density field solution has the following form:

where the reference value of densitiesand are calculated through use of a thermodynamic equation of state (EOS) by using the molecular weights of heavy and light fluids ( The characteristic initial mixing length-scale is denoted by . The mixture fraction field is now defined as,

Using the one-dimensional continuity equation gives the wall-normal component of velocity as,

It should be noted that all the above expressions have a singularity at time (*t =0*), thus an offset time, , is added based on diffusion time scales to avoid the singularity. The analysis of this problem begins at =10. Also note that the reference density fields corresponding to isothermal light and heavy fluid mixing in air are defined through the use of the equation of state (Table 1). The species fraction of oxygen and nitrogen in air are: and , they are used calculate the molecular weight of the mixture composition,

and

The verification task begins by initializing the code with the above equations at an offset time,=10. The domain is selected as *(Lx, Ly, Lz) = (10,30,5)* with periodic conditions in *x* and *z*. Grid design procedures are followed in order to resolve the characteristic mixing layer which has a minimum length-scale of 2 and increases to 20 by the end of the simulation. Adequate resolution will ensure to have at least 30-40 points inside the mixing layer thickness set by the initial conditions. The running time is selected based on an analysis off the time evolution of the minimum value of mass density (Figure 2) which shows that the peak mass density decreases exponentially to nearly 1 at an approximate time of 400 time units.

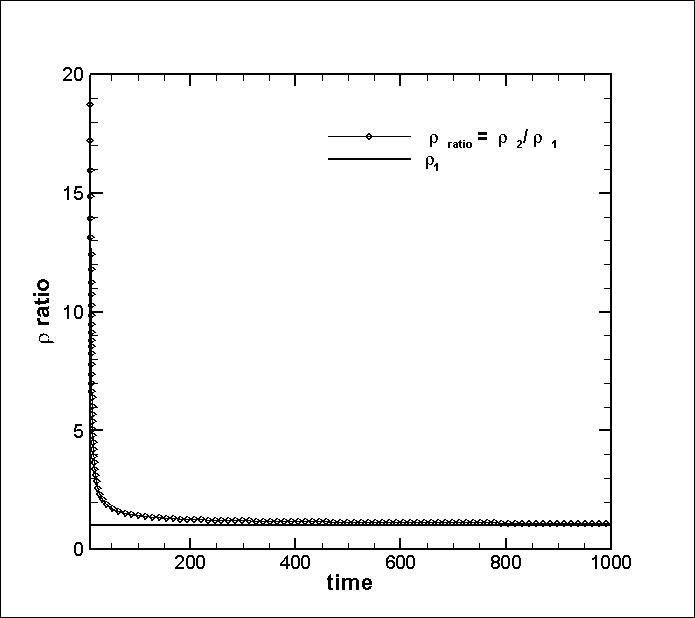


Figure 2. Time variations of minimum value of mass density in binary mixing solution.

|  |  |  |  |  |
| --- | --- | --- | --- | --- |
|  |  | **(off-set time)** |  |  |
|  |  | 10 |  |  |
|  | **(characteristic lengthscale)** |  |  |  |
|  | 50 |  |

**Physical Parameter Table**

Table 1. Parameter Table, 1D mixing case.

**Computational Model Table**

|  |  |  |
| --- | --- | --- |
|  |  |  |
| (10,30,5) | (10,1200,4) | Uniform grid in *x, z, y* |
| **Initial Conditions** | **Specific Enthalpy model** |  |
|  | Chemkin coefficient data base w/  enthalpy polynomial: |  |
|  |  |  |
|  | QUICK | Spanwise, Streamwise |

Table 2. Model Parameter Table, 1D mixing case.

Figure 3. Mas-density, velocity and mixture fraction profiles at time *t =10*.

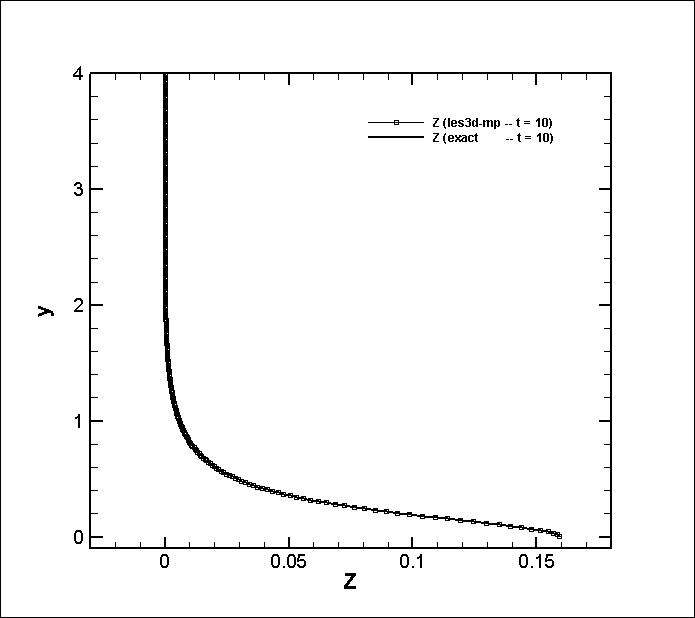
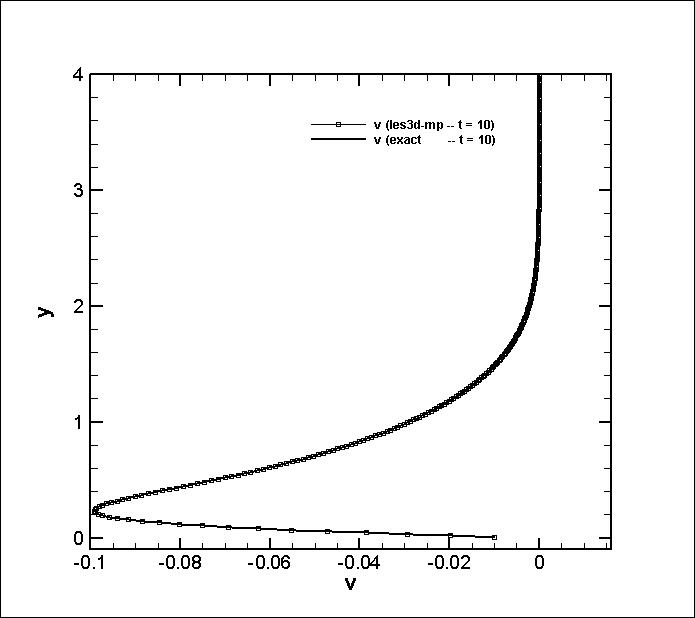
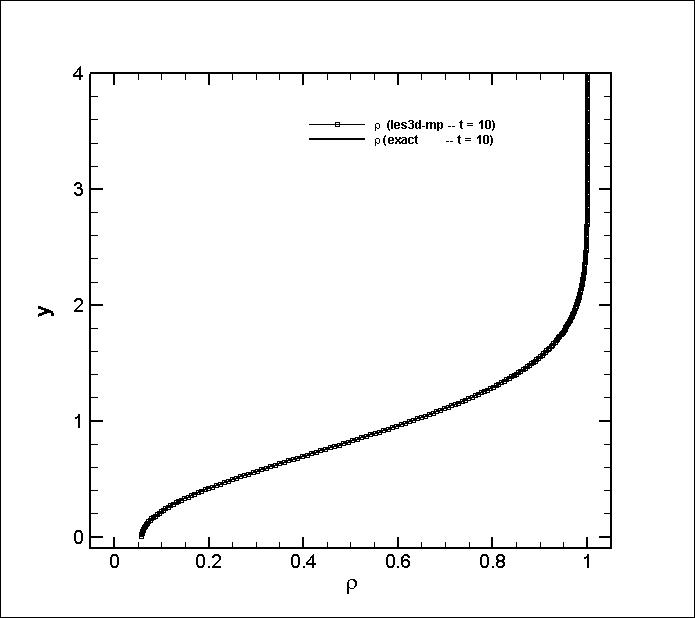
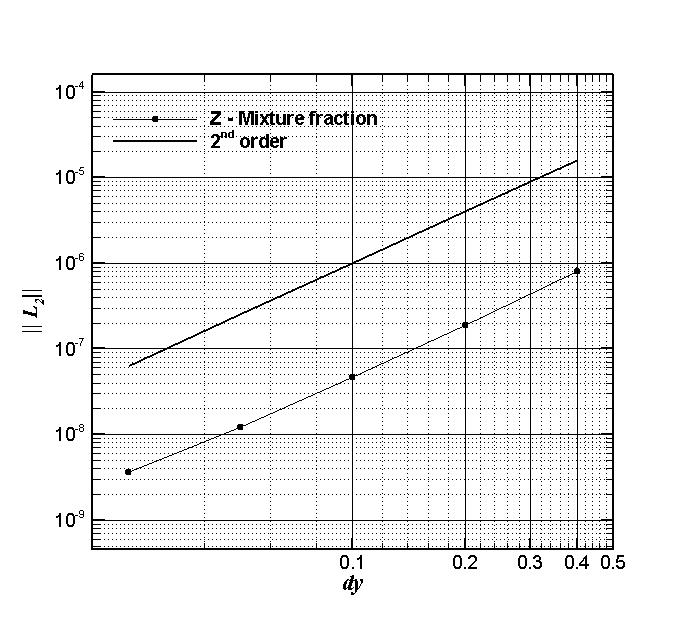
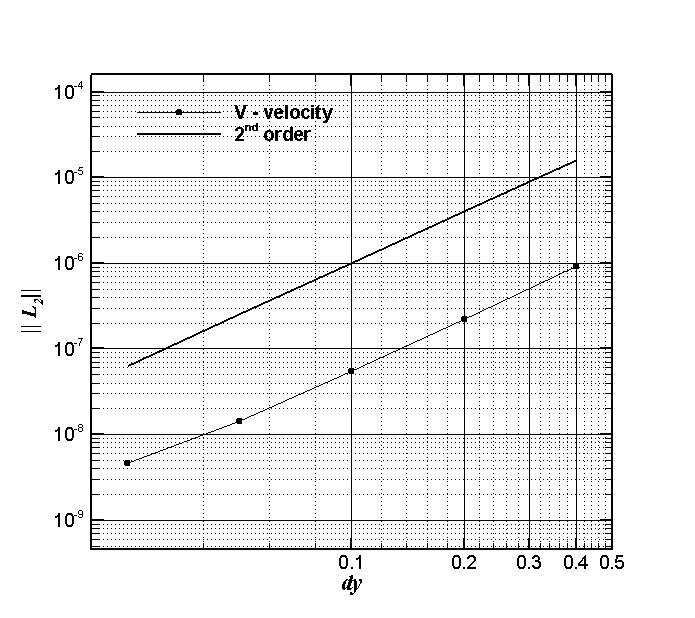
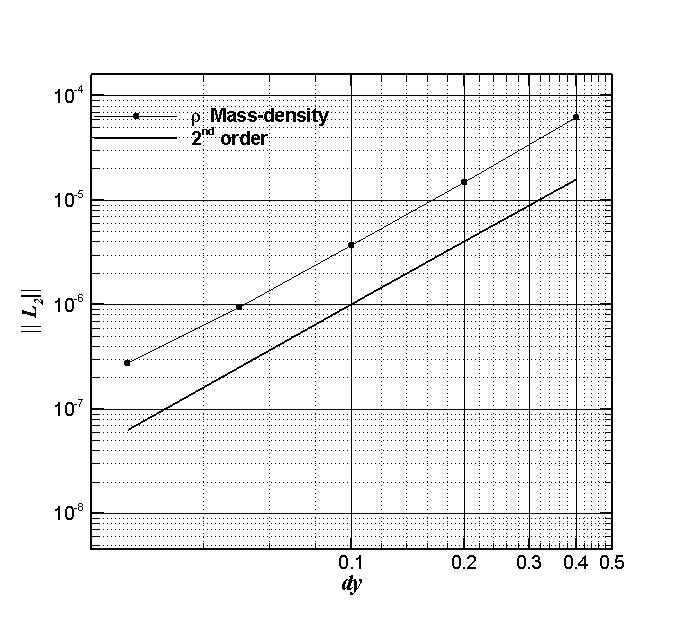


Figure 4. Spatial order of accuracy analysis, 1D binary mixing problem calculated at t = 1000 time units.



The case study was setup with the physical parameters presented in Table 1 which specify the reference Reynolds number and Prandtl number as 50 and 0.75 respectively. Initial density limits and characteristic length and time scales are also presented. The computational configuration, including domain size and resolution are presented specifying the wall-normal grid-spacing value of . Initial conditions are prescribed from the analytical solution at a prescribed offset time while also enforcing the boundary conditions. Figure 3 shows instantaneous profiles of mass density, y-velocity and mixture fraction at the reference time, . As was stated earlier this problem corresponds to isothermal mixing between two fluids of different molecular weights. This can be observed from the mass density profile in figure 3 showing a clear density mixing layer. The negative y-velocity component indicates the direction of the fluid motion and has maximum values at regions with peak density gradients. All profiles presented on figure 3 compare very well to the analytical solution.

Figure 4 shows a spatial order of accuracy analysis for profiles of mass-density, y-velocity and mixture fraction. The analysis is performed by choosing a fixed time step of and refining the grid with respect to the y-direction. Note that the time step, *dt*, has to be small enough to provide an analysis that is not contaminated by temporal discretization errors. The number of points used in the refinement studies are: and the global error, , is used to show the observed order of accuracy of the finite differencing scheme (where is the exact solution and is the discrete solution and *N* is the number of grid points). As indicated by Figure 4 the second-order accuracy of the code is retrieved in the presence of flows with strong density gradients.

**3.3 Temporal accuracy cases**

**3.3.1 Homogenous case**

This section deals with showing the order of accuracy of the time integrator in les3d-mp. Two time integrating options are available; they are second order Adams-Bashford and the classical third order, low-storage Runge Kutta scheme presented in section 2. A simple transient problem strictly defined as a function of time only, *f(t)* is selected in order to remove spatial errors from analysis. This takes the form of,

where and with the following initial condition, . Since this test case is independent of spatial variables it can be run on a very coarse computational grid, such as *(nx,ny,nz) = (4,4,4)*, or 43 and with a Reynolds number = 1. It makes use of periodic conditions in spanwise and streamwise directions as well as velocity symmetry conditions at the wall-normal boundaries. Symmetry conditions are defined as zero gradients for *u* and *w* velocities and zero value for v-velocity.

The velocity oscillation is introduced into les3d-mp by advancing the mean pressure gradient in time as the right hand side of the integrated solution. It is described through the mean pressure equation, , where the pressure gradient amplitude, *A*, and the wave period, *T*, are given in Table 1. This transient ODE has the following analytical solution that can be used when performing temporal order of accuracy studies,

Table 3. Parameter values for transient test case.

|  |  |
| --- | --- |
| **parameter** | **value** |
| T | 10 |
| A | 1 |
| Re | 1 |
| nxnynz | (4,4,4) |
| Periodicity | x, z |

The order of accuracy analysis is performed by specifying a grid resolution and refining the time step sequentially. A total simulation run time of 1.5 time periods or 3*T*/2 was selected. Since this is a temporally homogenous case the spatial resolution prescribed by the grid does not play a role in the analysis. The number of points used for each refinement study were: and the definition of the global velocity error, , is used to show the observed order of accuracy of the time advancement scheme (where is the exact solution and is the discrete solution and is the number of time steps). As presented in Figures 5 and 6 this establishes the 2nd and 3rd order accuracy of the Adams-Bashford and Runge-Kutta scheme.

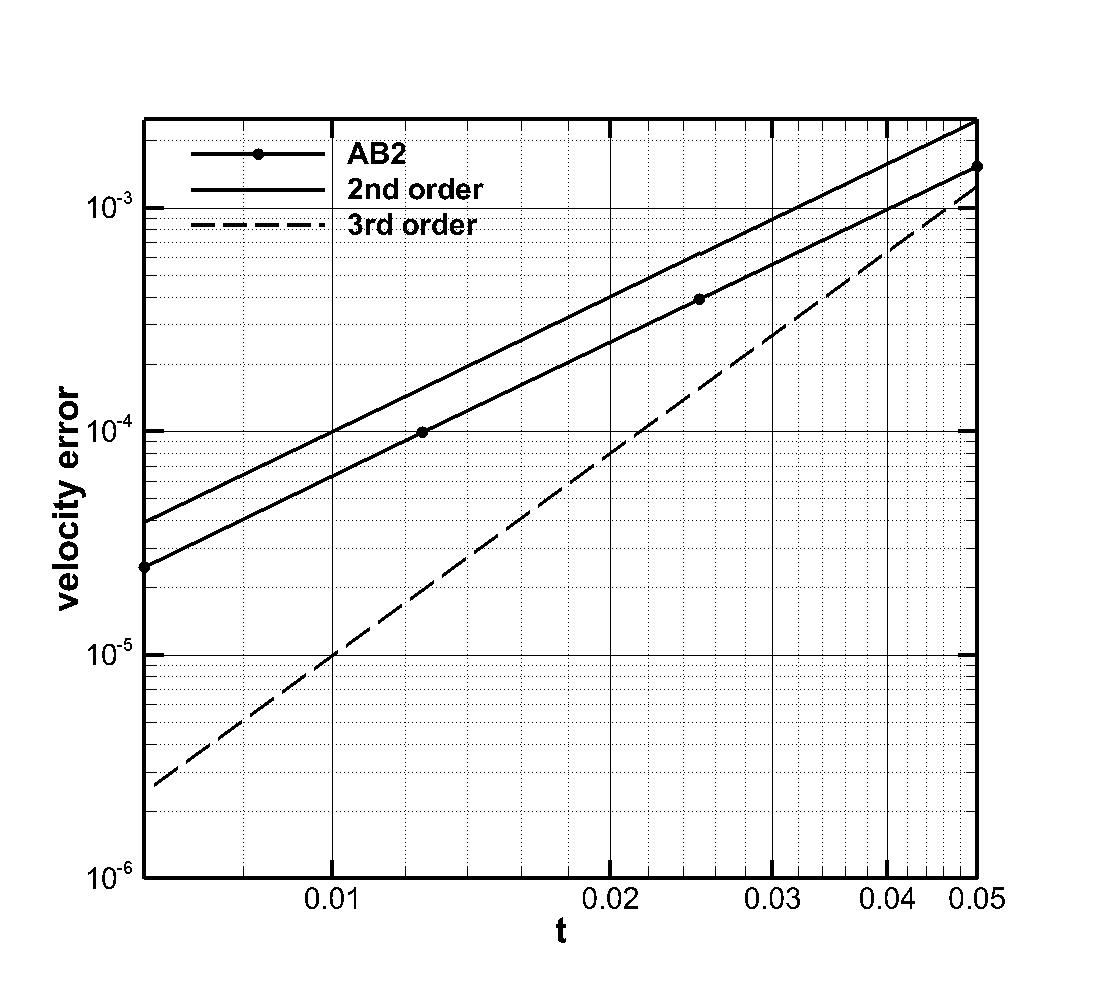


Figure 5. Order of accuracy analysis for AB2.

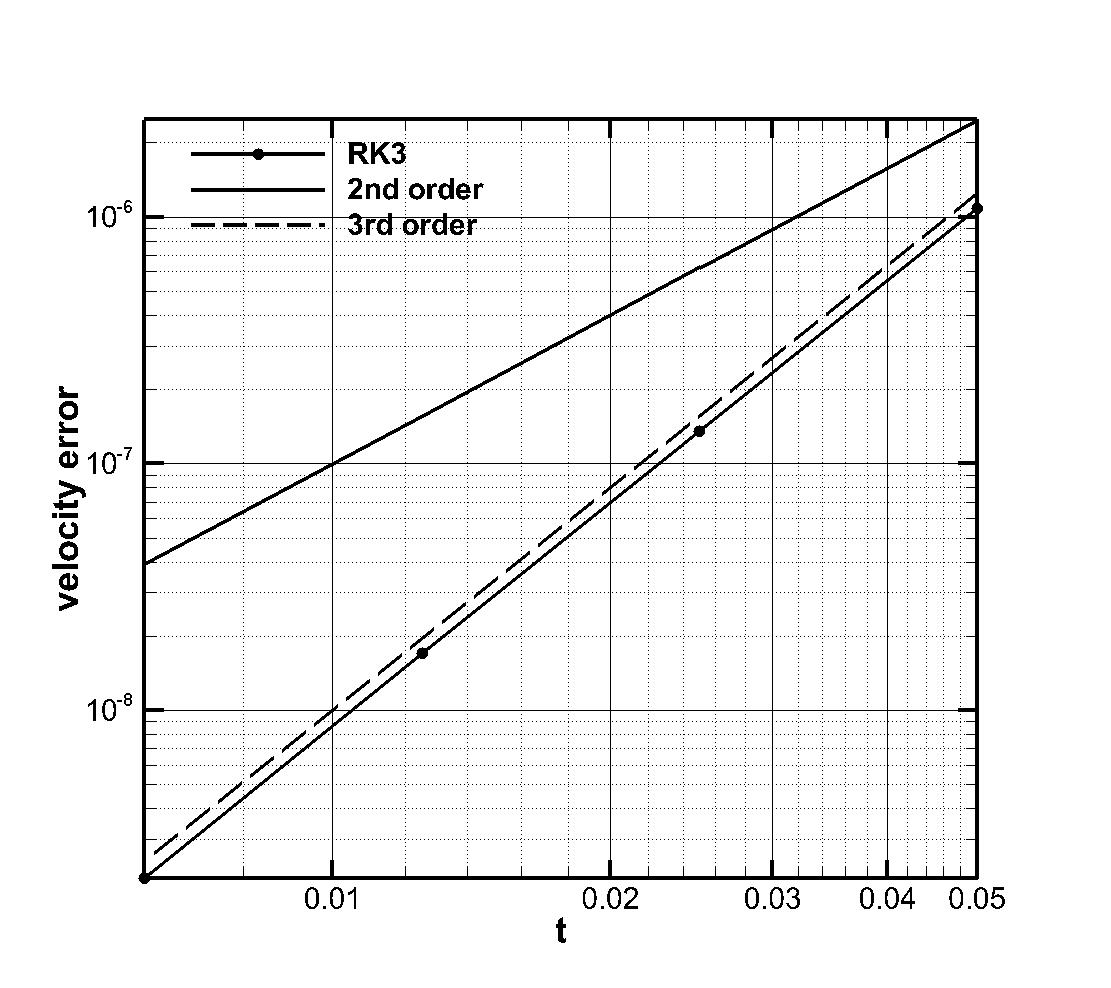
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Figure 6. Order of accuracy analysis for RK3.

**3.3.2 Time-Periodic Poiseuille**

This section presents an order of accuracy analysis for the time-periodic Poiseuille flow. This variation of the classical flow brings in unsteady velocity features which are specified through an oscillating pressure gradient. The unsteadiness provides regions of local acceleration/deceleration and regions of flow reversal making this an interesting academic problem. The purpose of the time-periodic flow is to show temporal order of accuracy of the code when the form of the equation being solved is dependent on both temporal and spatial constituents, i.e., . This provides a useful verification test of the time integration scheme where the solution depends on spatial and temporal variables. The axial pressure gradient is here expressed, via Fourier series, in terms of sinusoidal functions of time which lead to a complete representation of velocity, pressure and wall-shearing stress associated with this flow. The harmonic pressure gradient is specified as , where is the standard angular frequency of the system, and represents the amplitude. (Note the pressure gradient can also be re-written, using Euler’s identity, as the sum of trigonometric harmonic functions).

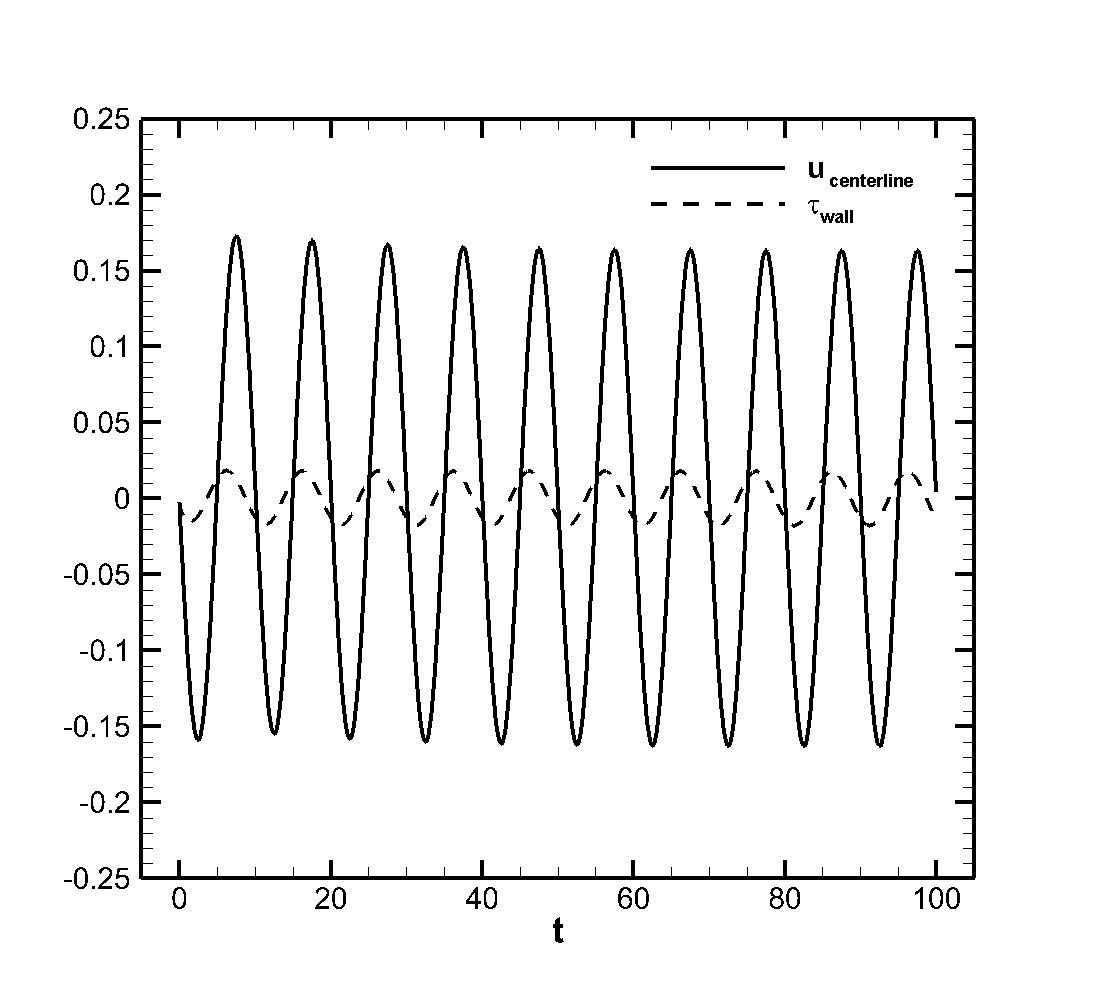
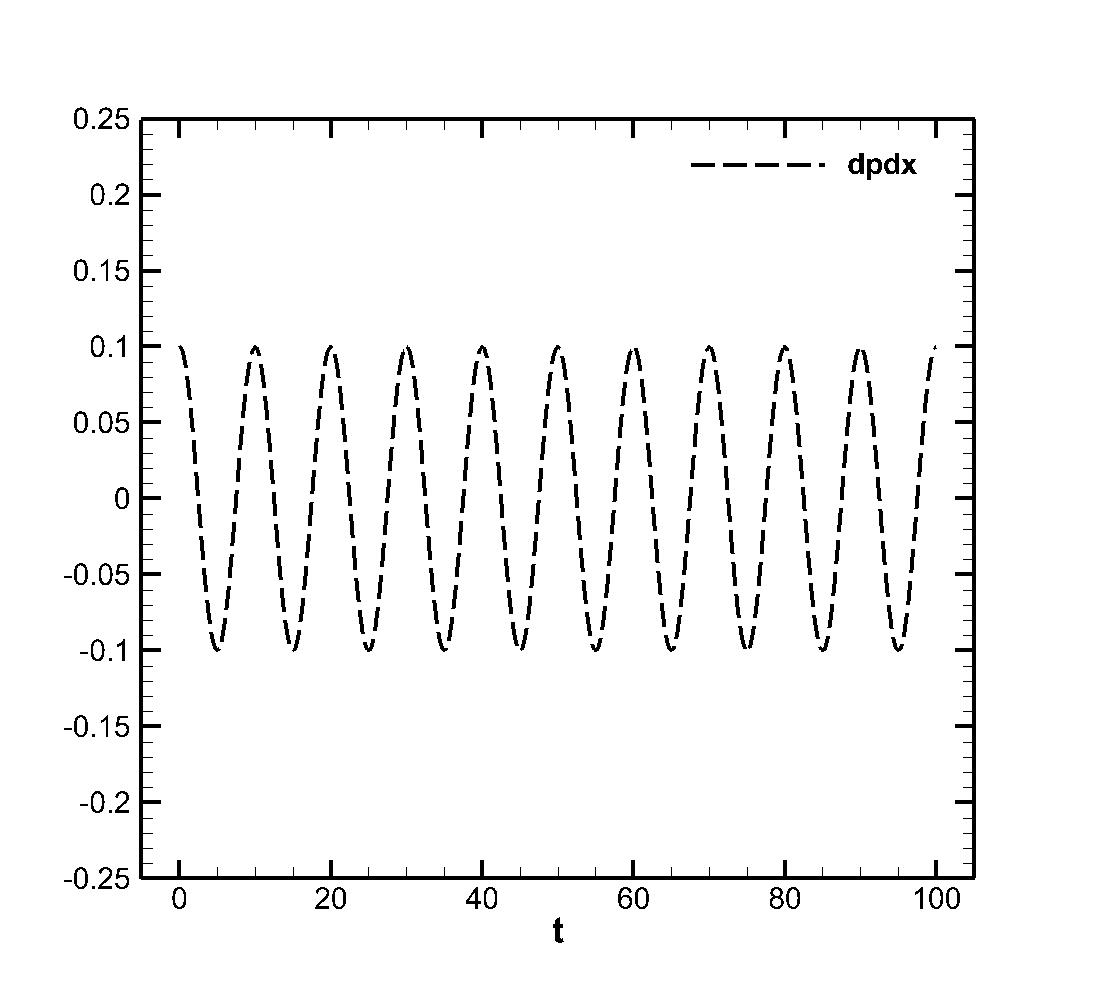


Figure 7. Time-periodic Poiseuille flow. Fourier axial pressure gradient (left), centerline velocity and wall shear stress profiles (right).

Figure 7 shows the oscillating pressure gradient along the streamwise direction with a specified period, and amplitude, . The flow is run for a total of 10 periods to smooth out initial transients and to allow the flow to adjust to its fully developed state. Figure 7 also presents peak centerline velocity values which show acceleration and deceleration regions inherent to pulsating flows. This pulsating phenomenon also introduces oscillations in wall shear stress which is clearly seen in the time history evolution. A series of finer resolution cases were studied in order to study the temporal accuracy using the error of the centerline velocity. The error was computed by comparing to a highly resolved reference solution. The results show 2nd order accuracy for the Adams-Bashford scheme. Numerical issues are still present with the integrations of RK3 scheme and the accuracy results are thus not shown in this present study.

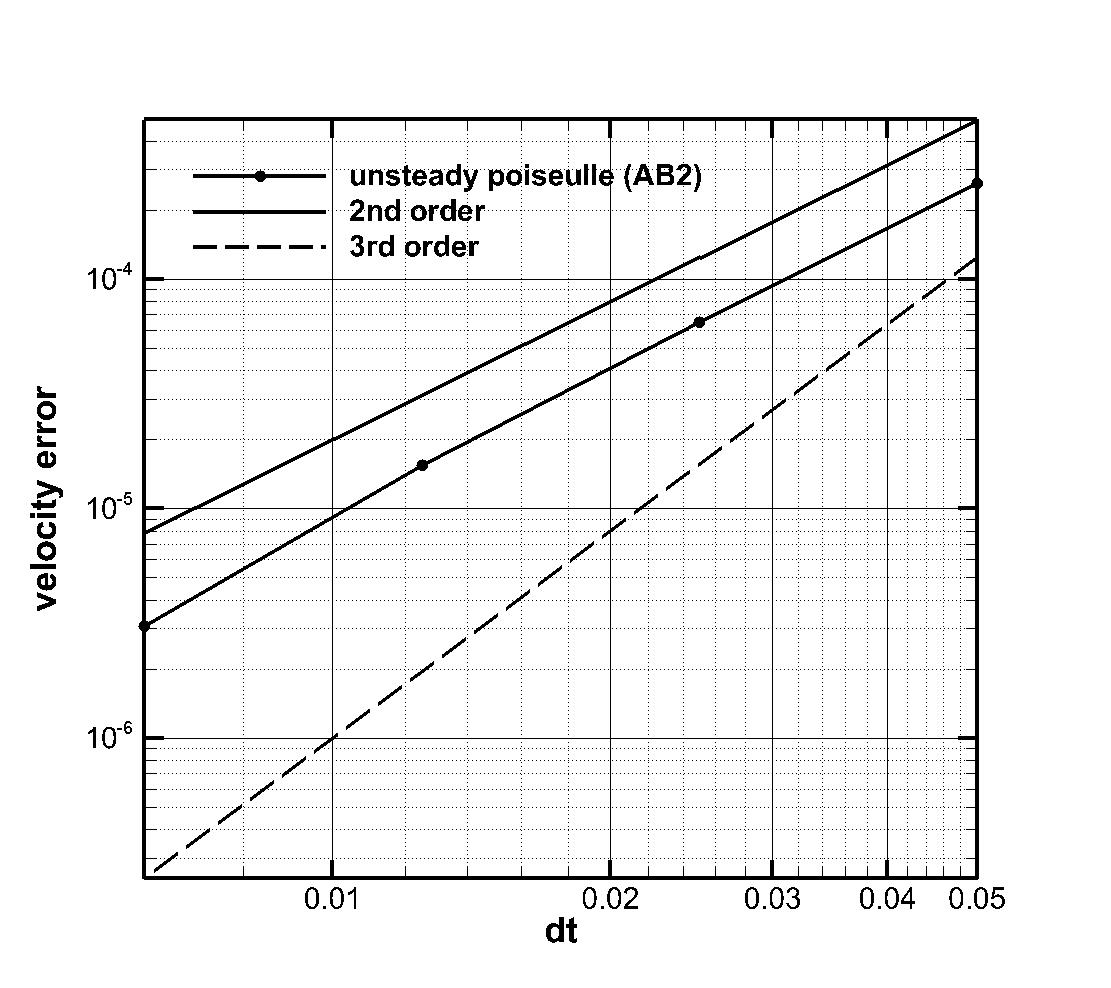
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Figure 8. Order of accuracy analysis for Time-Periodic Poiseuille flow.

**3.4 Classical boundary layer solutions**

**3.4.1 Forced convection boundary layer: Blasius Solution**

We now focus our attention to modeling a Blasius flow configuration with and without heat transfer. Physically this is an important test case since it captures some of the salient features (i.e, wall shear stress, heat flux) found in momentum-driven boundary layers. Numerically this is also a challenging flow since it requires the implementation of inflow and outflow conditions to capture the spatially developing nature of the problem. The parameters necessary to implement this run are presented in Table 1 and results are shown for representative cases.

**3.4.1.1 Results**

The physical parameters for the Blasius test case are presented in Table 4. The Reynolds number is the critical non-dimensional number controlling the physical properties and state of the flow. It is defined with the displacement thickness, , as the reference length-scale and a free-stream velocity of unity as the velocity-scale so that, . The Prandtl number is specified according to its reference value in air. The simulation is performed on a computational domain of *(Lx, Ly, Lz) = (200, 60, 5)* with the first off-wall grid point located at and the spacing of the outer points scaling with the boundary layer thickness, (The total number of grid cells is 50,000). Standard grid design procedures are followed ensuring that 30-40 points are used to resolve the boundary layer viscous region. This is achieved through grid stretching by means of a hyperbolic tangent function which clusters grid nodes near the wall. This capability allows us to be more efficient in resolving the boundary layer thickness, .

**Physical Parameter Table**

|  |  |  |  |  |
| --- | --- | --- | --- | --- |
|  |  |  |  |  |
|  |  |  |  |  |
|  |  | *(displacement thickness)* | *(boundary layer thickness)* |  |
|  | 842-10,000 |  | 3 |  |
|  |  |  |  |  |
| 1.0 | 1.0 | 0.02 | 50 |  |

Table 4. Parameter Table, Blasius case.

**Computational Model Table**

|  |  |  |  |
| --- | --- | --- | --- |
|  |  |  | |
| (200,60,4) | (128,96,4)  0.028 | * Wall –normal hyperbolic   (= 2.75)   * Uniform grid in x, z | |
|  |  |  | |
| (where the theoretical Blasius profiles are specified) |  | 0 | *(Option)* |
|  |  |  | |
|  | CHEMKIN coefficient data base for  enthalpy polynomial: | CHEMKIN coefficient data base for  heat capacity polynomial: | |

Table 5. Model Parameter Table, Blasus case.

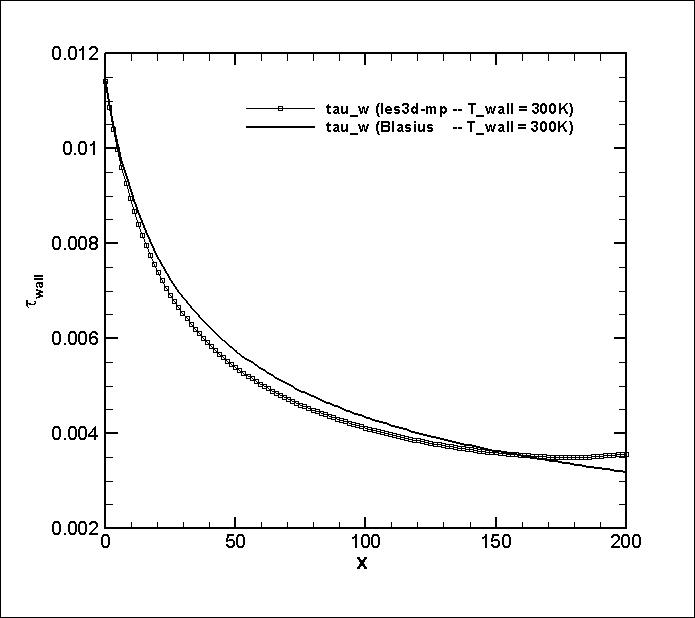
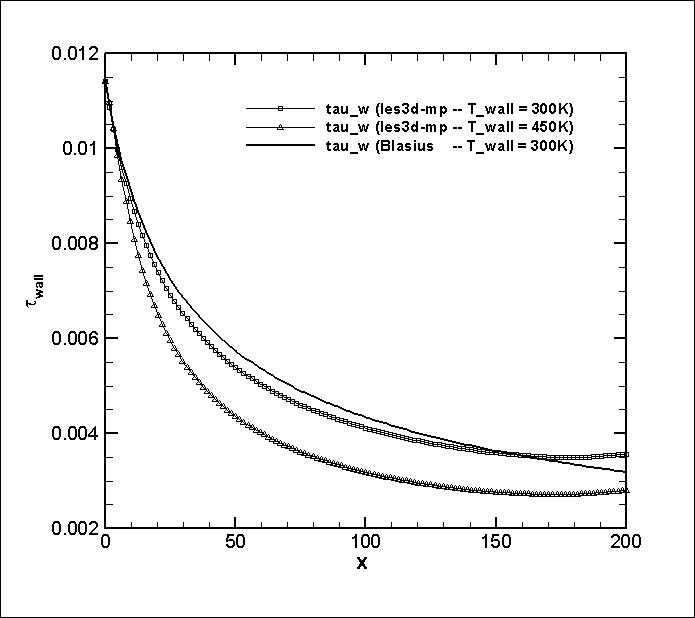


Figure 9. Blasius wall –shear stress no heat transfer (top), comparison to case with heat transfer (bottom).

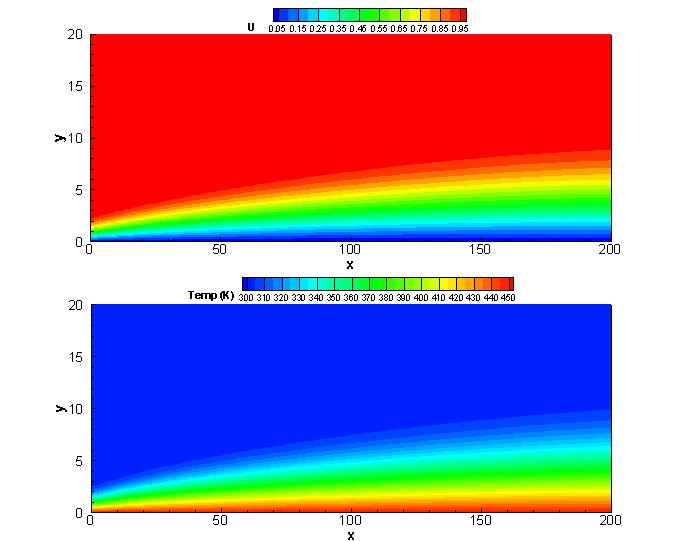
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Figure 10. Velocity and temperature contours of Blasius case with variable density heat transfer.

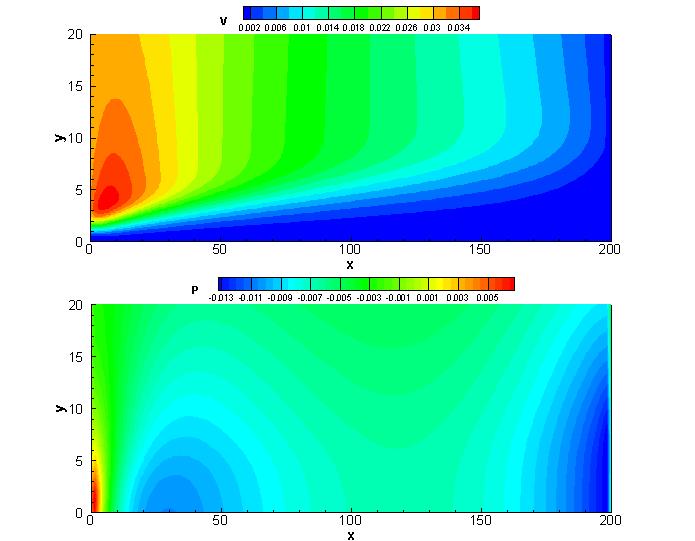
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Figure 11. Lateral velocity and pressure contours of Blasius case with variable density heat transfer.

The specification of the inflow profile for the constant density boundary layer case is specified through the use of Blasius profile as is shown in Table 5. Blasius *u* and *v* velocity profiles to be used as inflow conditions can be prescribed by specifying the reference viscosity and the displacement thickness into a simple auxiliary program that is called inside the inflow module. A more elaborate inflow condition is specified for the case with heat transfer. This makes use of the similarities between the momentum and total enthalpy transport equation in Blasius type flows where the effect of pressure gradient is negligible, . Using this concept one can write the following description for the enthalpy inflow condition as where the values for enthalpy at the free-stream and at the wall are defined using the corresponding temperature (and mixture fraction, for air cases).

Figure 8 shows the spatial evolution of the wall-shear stress. The top figure shows good agreement of the wall shear stress with the Blasius exact solution. Comparisons of the wall-shear stress to the analytical expressions have to account for distance from the leading edge distance which is a virtual origin and is not shown in the computational domain and is implicitly defined in the inflow profiles. Thus, the reconstruction of the wall-shear stress profiles are calculated through the modified analytical expressions, where the inflow Reynolds number contribution is accounted for by the expression, (and the leading edge distance is, ). An additional figure is shown in Figure 9 corresponding to a case with heat transfer. The effect of heat transfer is seen to reduce the magnitude of the wall velocity gradients, this is turn decreases the wall-shear stress.

Figure 10 shows contours of velocity and temperature for the case with heat transfer at a wall temperature of . As is typically observed for momentum driven flows the growth of the boundary layer scales like (or ). This behavior can be directly observed from the velocity and temperature profiles as the increases (or distance along the plate). Figure 11 shows contours of wall-normal velocity and pressure contours in the flow. Although not severely adverse, errors associated with the approximate boundary conditions at the outflow can be seen through the magnitude of the y-velocity at that location (To obtain a better result the outflow boundary should include wall-normal diffusion terms which are inherent to Blasius-type flows). At the inflow an incorrect initial region with peak y-velocity magnitudes arises due to the approximate specification of a Blasius profile, which is restricted to a constant density framework. Similar to the observations made for the y-velocity plot, the pressure contours show peak disturbances near the inflow and the outflow. Aside from the pressure disturbances, the pressure variations remain small, three order of magnitudes lower that the v-velocity component and one order of magnitude lower than the u-velocity.

**3.4.2 Natural convection boundary layer: Ostrach Solution**

The solution of a laminar free convection boundary layer next to a heated isothermal plate was derived and developed by Ostrach [8]. This analysis presents a quasi-analytical solution to the problem based on similarity arguments and coupling between the temperature and velocity fields. This is an interesting test case for our purpose because it involves the modeling of a buoyancy driven boundary layer flow often found in applications involving fires. The modeling involves a vertical isothermal hot wall at 360 K surrounded by an environment at   
300 K. That difference in temperature generates a laminar natural convection flow-structure in the fluid. This is shown in the following figure,

Figure 12. Free convection boundary layer schematic.

**Physical Parameter Table**

|  |  |  |  |  |
| --- | --- | --- | --- | --- |
|  |  |  |  |  |
|  |  |  |  |  |
|  |  |  |  | **­** |
|  |  |  |  |  |
|  |  |  |  |  |
| 1.1 | 2.0 | 0.02 | 110 | 16.2 |

Table 6. Parameter Table, Ostrach case.

**Computational Model Table**

|  |  |  |
| --- | --- | --- |
|  |  |  |
| (10,10,4) | (100,60,4) | * Wall –normal hyperbolic   (= 2.75)   * Uniform grid in x, z |
|  |  |  |
|  |  |  |
|  |  |  |
|  | CHEMKIN coefficient data base for  enthalpy polynomial: | CHEMKIN coefficient data base for  specific heat polynomial: |

Table 7.Model Table, Ostrach case.

The table above presents the parameters used in running the test cases in les3d-mp. The simulation was performed for a larger than normal viscosity value of . Due to the Grashof number dependence on the viscosity and its critical value needed for transition to turbulence ( we can see that the effect of increasing the viscosity is also to increase the spatial region in the laminar regime. The set-up of this case in les3d-mp involved prescribing the temperature of the hot wall (360 K) and allowing the flow to develop naturally. The reference values used in the run were taken at an elevation height of defining the Reynolds and Froude number shown in table.

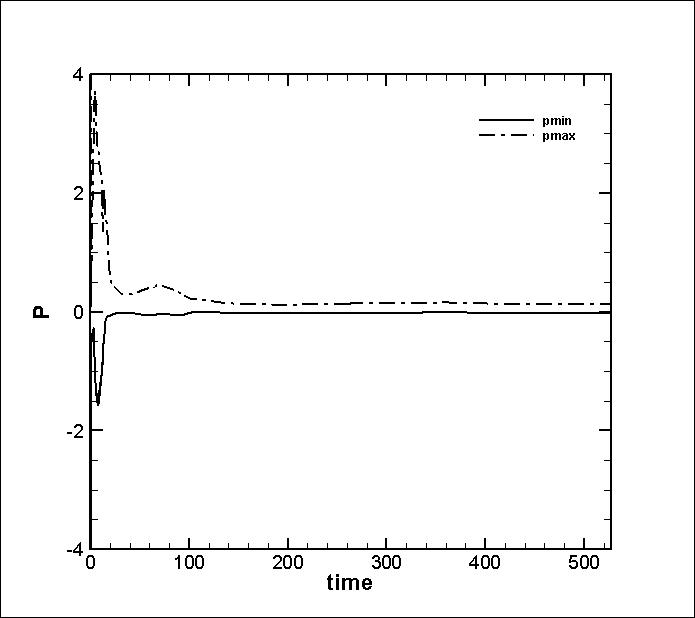
****The simulation was run in lesd3-mp for a total of 500 time units, or 50 flow-through times defining a flow-through time as an estimate of the time it takes for a fluid particle to travel through the length of the domain. A good indication of the quality of the solution is the small magnitude of the peak pressure oscillations since unphysical pressure surges will adversely influence the flow. The following figure shows minimum and maximum pressure perturbations with large values at initial times due to the adjustment of the flow.

Figure 13. Time history of peak pressure.

**Heat Flux**

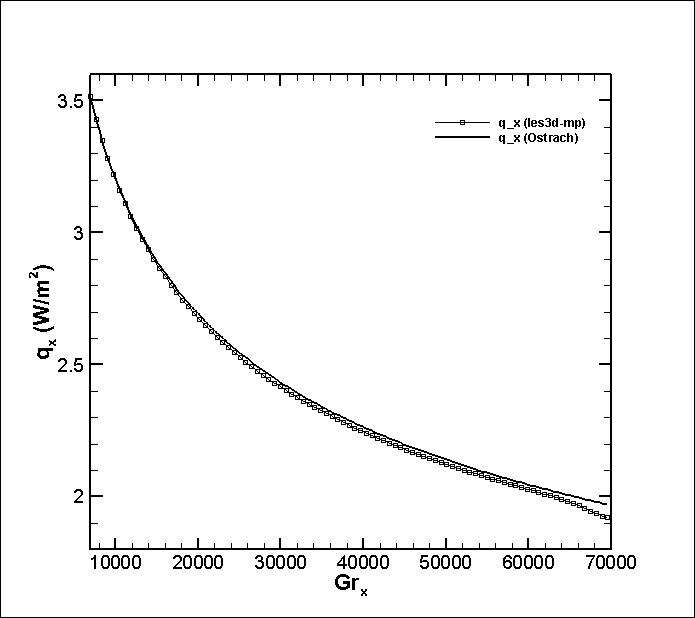
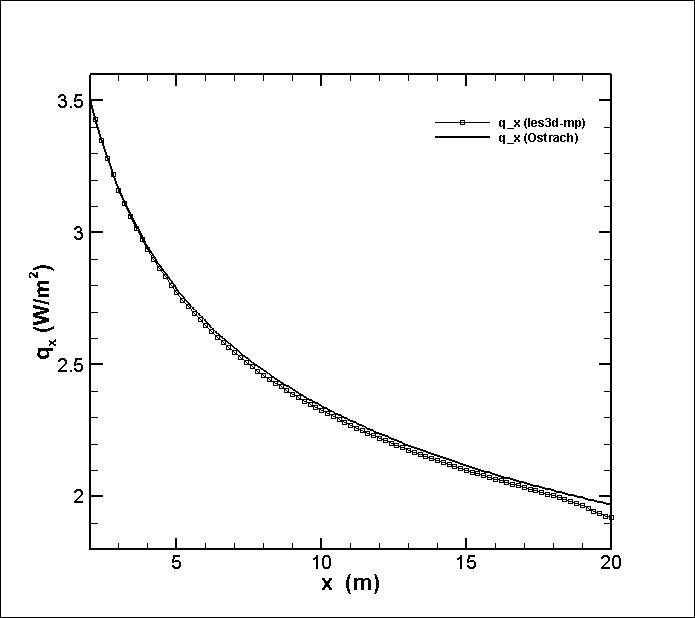


Figure 14. Heat flux comparison to Ostrach solution. Comparison with respect to plate elevation (top), comparison with respect to Grashof number (bottom).

**Nusselt**

Figure 15. Nusselt number comparison to Ostrach solution. Comparison with respect to plate elevation (top), comparison with respect to Grashof number (bottom).

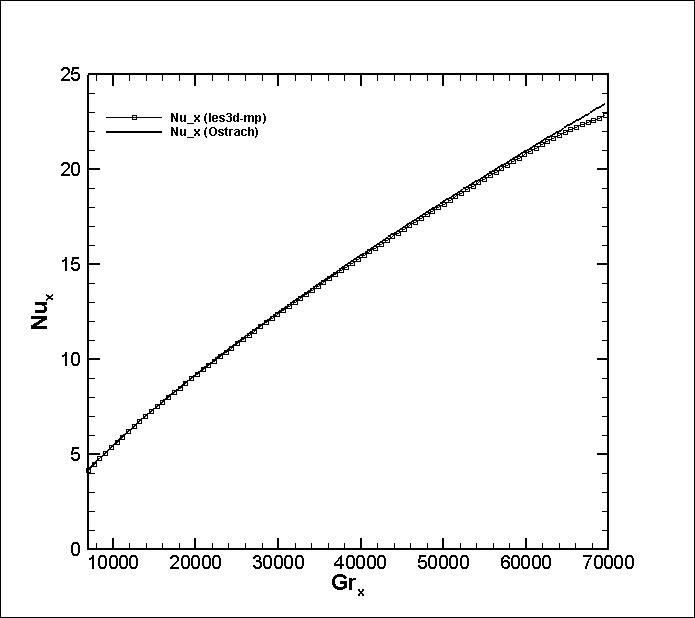
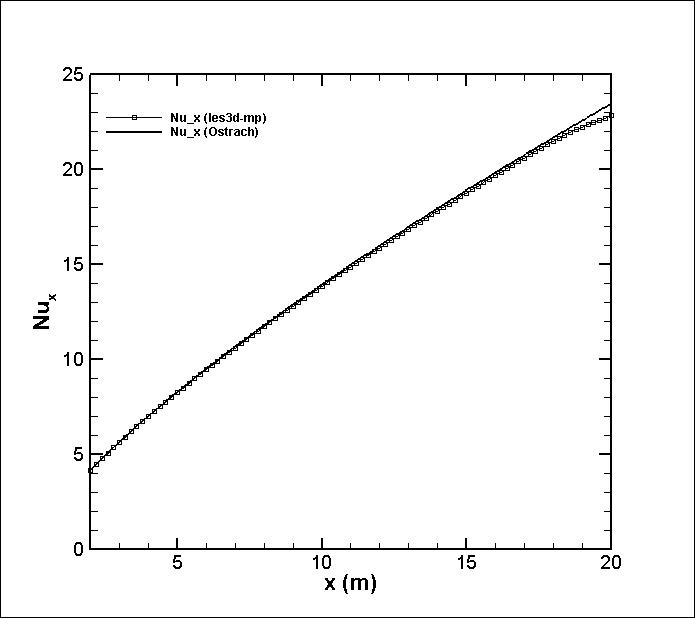


Figure 16. Velocity profiles at elevations x= 2m, 3m, 3 m and compared to Ostrach.

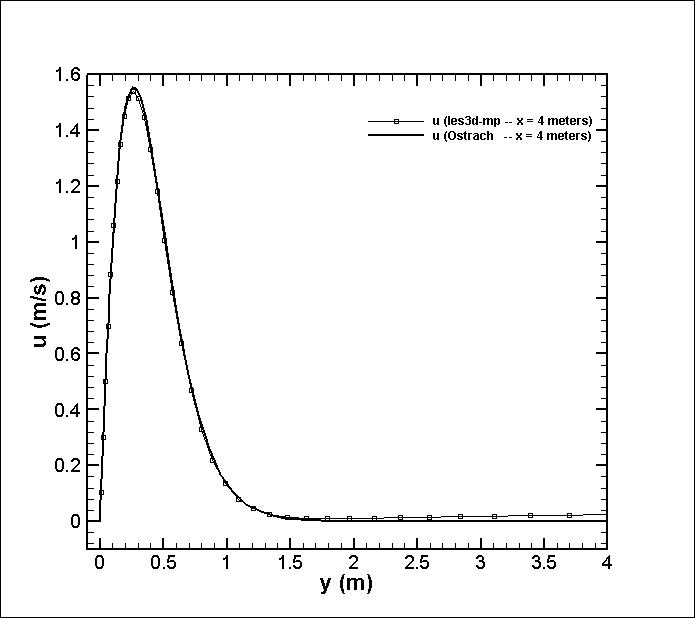
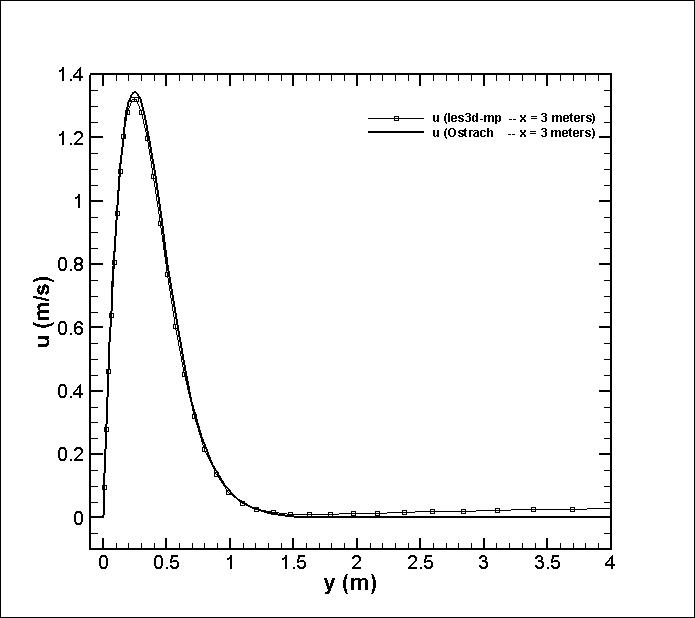
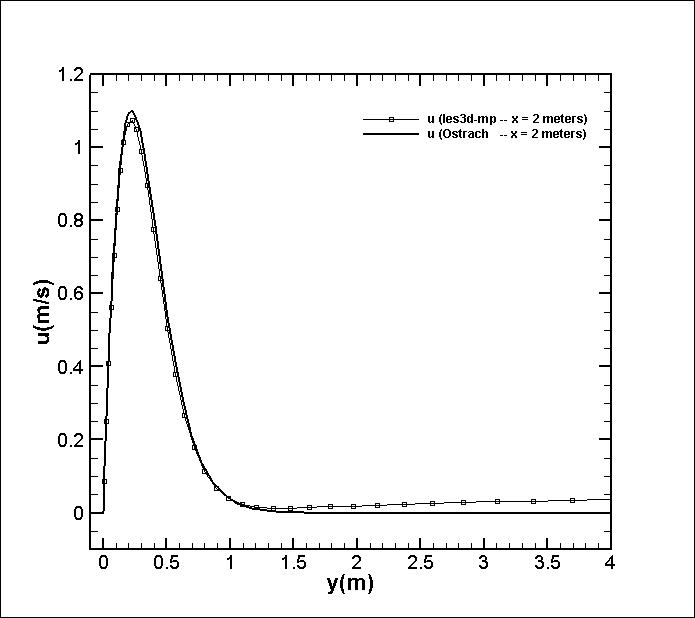


Figure 17. Temperature profiles at elevations x= 2m, 3m, 3 m and compared to Ostrach.

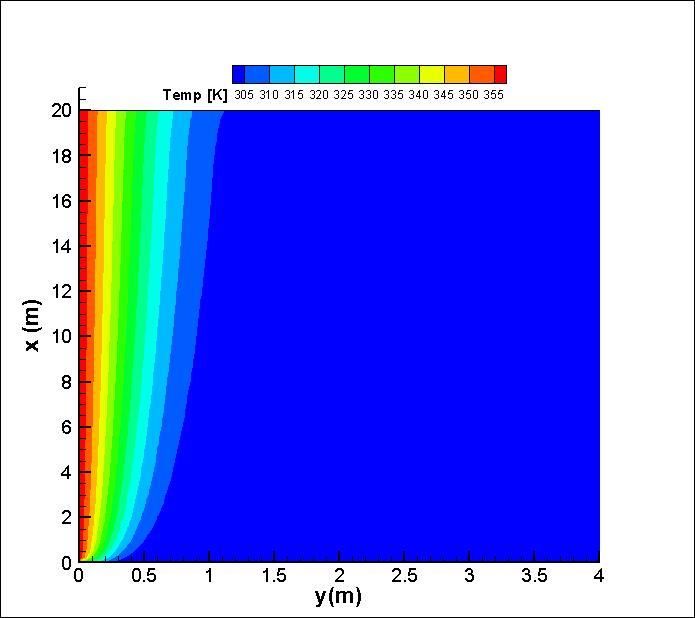
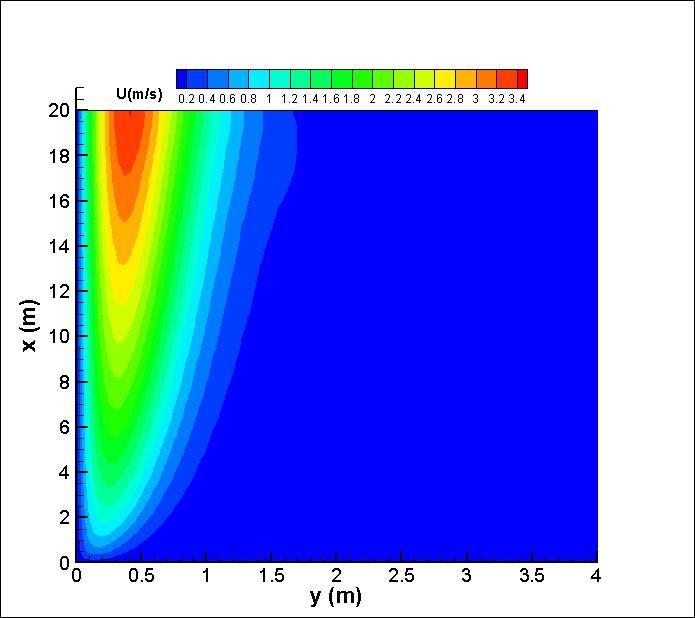
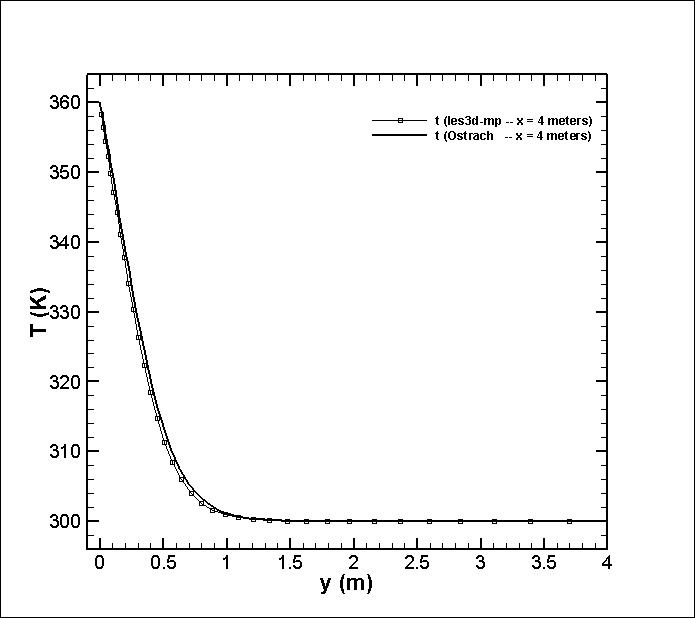
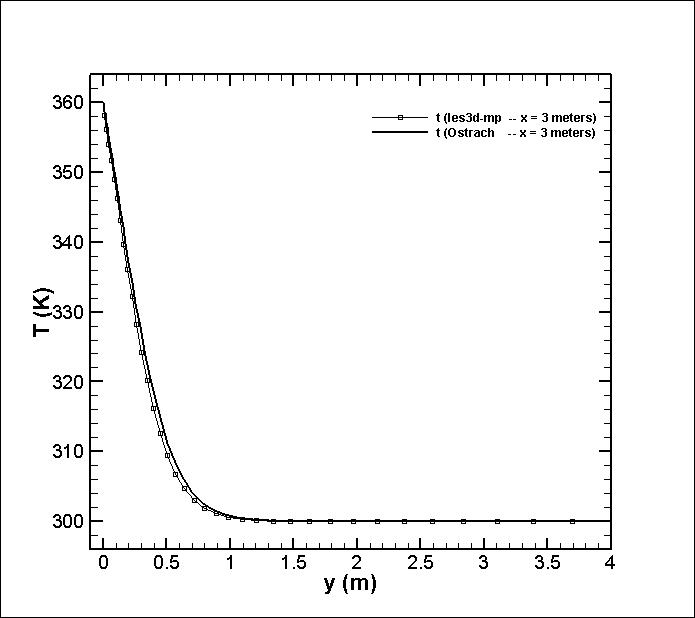
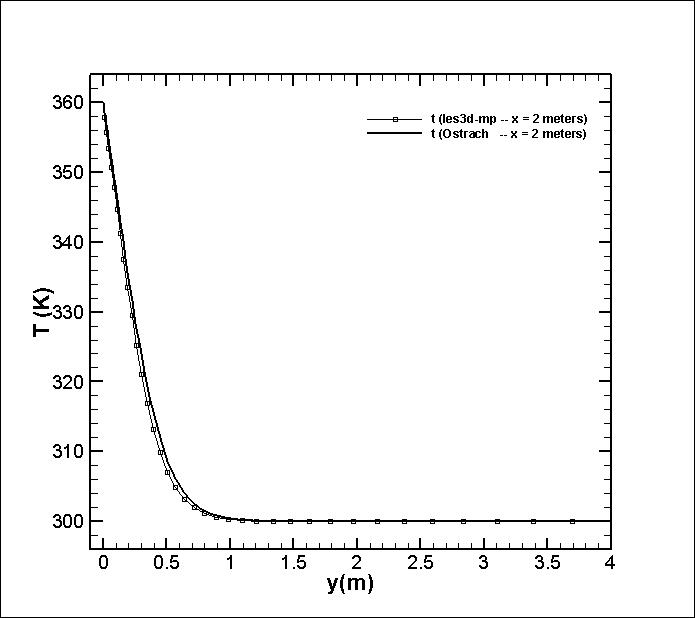


Figure 18. Natural convection u-velocity and temperature contours.

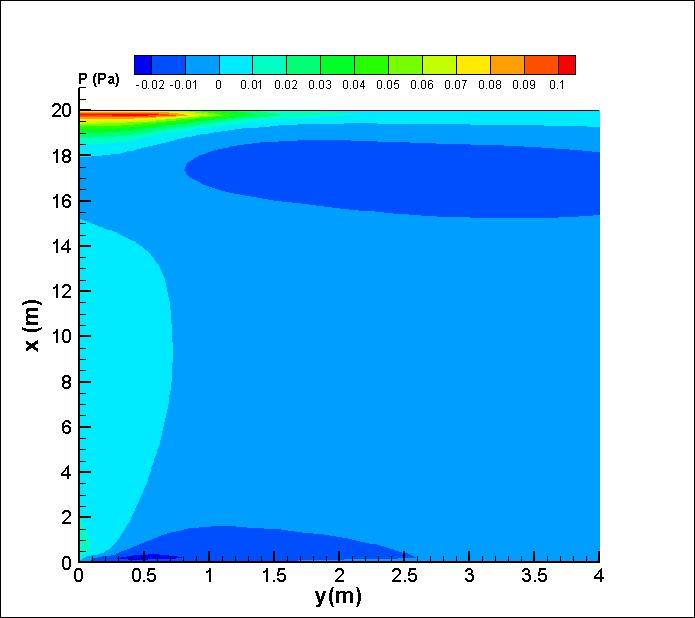
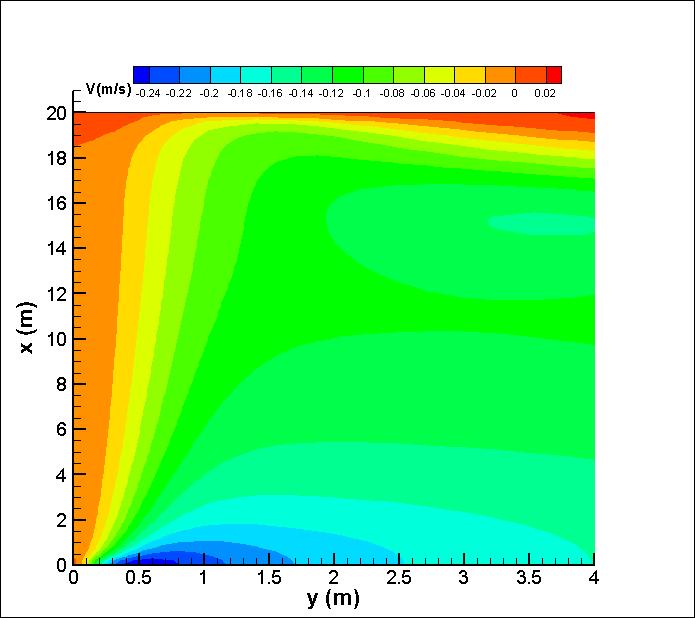


Figure 19. Natural convection pressure and v-velocity contours

The results presented in figures 14-15 show good agreement with the theoretical Ostrach solution. The heat flux is presented as a function of both elevation and the Grashof number; this shows the spatial development of the solution (In free convection flows, the heat flux varies like, ). Similarly, Nusselt number results show close comparison to the analytical solution establishing the quality of the heat transfer calculation. We have also noticed problems near the outflow. This occurs starting at ~ 18 meters to 20 meters and the peak deviation is less than 1 %. Although small, this represents a problem in the outflow boundary conditions which can influence more sensitive diagnostics such as the entrainment velocity. This can be seen from the wall-normal velocity contours on Figure 18 where an unphysical surge of v-velocity is seen near the outflow. Figures 16-17 show profiles of streamwise velocity and temperature at three reference stations. The reference stations are 2, 3 and 4 meters along the height of the plate. This shows good comparison to the Ostrach case inside the boundary layer thickness region with slight discrepancies of less than 2 %. Outside of the boundary layer region, the velocity profiles show the influence of a slight co-flow that was introduced at the inflow. The purpose of the co-flow is to control the streamwise velocity in regions where the flow is quiescent (such as the outside the boundary layer region) due to difficulties with Orlansky boundary conditions. This type of ad-hoc methods are widely used in CFD in addition to more popular buffer regions (sponge layers) near the outflow to smooth-out outflow perturbations.

Streamwise velocity and temperature contours presenting the structure of the velocity and thermal boundary layer are shown in Figure 18. The classical boundary layer growth scaling can be retrieved by noting that the Grashof number, , plays the role of in buoyant flows and deduce that the boundary layer thickness, , is proportional to ­ (or ). This controls the development of the boundary layer. Figure 19 shows the pressure and v-velocity contours. The small magnitude of the pressure variations throughout the domain indicates a stable and accurate solution (as was also be seen by the peak pressure history). However, as was found even for momentum driven boundary layers, the use of approximate boundary conditions at the outflow introduces pressure disturbances that affect the quality of the solution. This can be directly seen by the alteration of the entrainment velocities near the outflow where the outflow model does not provide an accurate representation of the flow at that position.

**Conclusion**

This chapter presents fundamental studies of the order if accuracy of our in-house solver, les3d-mp. It includes a suite of verification tests that is essential to code developers in retrieving errors and establishing confidence in the code. The spatial accuracy test was done through the study of an isothermal mixing problem restricted to transport in one-dimension at large mass-density ratios. Computing the errors with respect to density, velocity and mixture fraction clearly revealed the second order accuracy of the central differencing scheme. Two studies were also performed to help temporal accuracy. The first study is a simple ODE which completely removes the effect of spatial errors and shows the correct behavior for our second and third order time-advancement schemes. The second study corresponds to the unsteady planar poiseuille channel flow problem. This is a valuable test since now the solution is a function of u-velocity and time: ; although not complete this study has shown successful second order of accuracy of the Adams-Bashford scheme. Lastly, classical momentum driven and buoyancy driven laminar boundary layer flows have been computed and compared to their analytical solutions: Blasius and Ostrach respectively. The results are good in the case of a Blasius configuration and satisfactory for the Ostrach configuration. The tests reveal that there is a need to improve the outflow boundary condition in problems that are buoyancy driven and regions where the flow is quiescent.

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