Concept of Operations

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Revisions

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1.0	Initial Release	10/30/2014	10/30/2014
1.1	Add software states to each software mode	10/31/2014	10/31/2014
1.2	Updated modes	11/23/2014	11/23/2014
1.3	Updated FSW flow chart	4/8/2015	4/8/2015
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Nomenclature

ADCS	Attitude Determination and Control Subsystem
AVI	Avionics Subsystem
BBB	BeagleBone Black
cFE	Core Flight Executive
CDH	Command and Data Handling
COM	Communications Subsystem
COTS	Commercial off-the-shelf
CSD	Canisterized Satellite Deployer
EPS	Electrical Power Subsystem
FSM	Finite state Machine
FSW	Flight Software
GPS	Global Positioning System
HM	Health Monitoring
NASA	National Aeronautics and Space Administration
NSL	Near Space Launch
PAY	Payload
RECONSO	Reconnaissance of Space Objects
RTD	Resistive Temperature Detectors
TCS	Thermal Control Subsystem
UHF	Ultra High Frequency

Summary

The RECONSO mission aims to detect and track space objects with the goal of demonstrating uncued space surveillance capabilities on a CubeSat platform with COTS hardware. This document outlines the Concept of Operations (CONOPS) of the mission by providing a day in the life outline of the mission as well as an outline of detailed operations on a per orbit basis.The purpose of this document is to guide software development to accommodate for the changes in mission architecture and execution. Two independent processes drive the operation of RECONSO. The FSW Health Monitoring (HM) process continuously monitors telemetry data from all subsystems and assesses if the entry criteria for each given mode is met. The process manager is an event-based manager that commands the operation of RECONSO using the predefined system operational modes outlined in this document. All of this control logic will be implemented using the NASA Core Flight Executive (cFE) open-source satellite flight software (FSW) architecture. cFE provides a number of built-in applications and a software messaging bus along which all of the given apps that RECONSO will use can communicate with each other. This document will outline each of the modes of operation of the satellite, detailing the entry and exit criteria of each mode and the states of all pieces of hardware and all blocks of software.

1   Overview of RECONSO hardware and software

Nominally, the RECONSO mission is accomplished by a camera and lens that take pictures of space objects as they pass across the field of view of the satellite. However, there are many other peripheral pieces of hardware and software that are required to operate this payload on-orbit. The table below reflects each of the pieces of hardware that will be flown on RECONSO. The Structures subsystem is not reflected in this table as it is assumed that their hardware will not be actively cycled or deployed as part of the CONOPS.

It should be noted that there are three different computers that are being flown on the RECONSO CubeSat. Their functions may be summarized as follows: The Tyvak Intrepid will run all FSW architecture and communications with the ground station. The Innoflight CFC-300 will be connected to the payload camera and will handle all image processing and payload operations. The BeagleBone Black (BBB) will be connected to all ADCS hardware and will handle attitude determination and control implementation.

The main blocks of RECONSO software are reflected in the following table. cFE allows for the packaging of various pieces of code as individual apps that run within the framework of cFE. Some of these apps come built into cFE, while others will be written by the RECONSO team to allow for more mission-specific functionality than what cFE is able to provide. It should be noted that there are also many pieces of software that are built into each one of these pieces of hardware by the manufacturers to allow it to function as intended.

Table 1: All components of RECONSO flight hardware and software. This table will be used multiple times throughout this document to highlight which pieces of hardware and software are operational in a given mode.

Subsystem	Hardware
PAY	Nocturn 50mm f/0.95 Lens
PAY	Nocturn XL CMOS Camera
PAY	Innoflight CFC-300 Processor
EPS	ClydeSpace 30Wr Battery
EPS	ClydeSpace PMAD Distribution board
EPS	In-house EPS Breakout board
EPS	In-house Inhibit board
EPS	In-house Solar panels
ADCS	Pumpkin GPSRM 1 GPS Receiver
ADCS	AntCom 1.5G L1 GPS Antenna
ADCS	In-house sun sensors
ADCS	In-house magnetorquers
ADCS	Tyvak Magnetometer
ADCS	BBB ADCS Controller
COM	ISIS UHF Dipole Antenna
COM	NSL GlobalStar Duplex
COM	NSL GlobalStar Patch Antenna
TCS	Omega Resistive Temperature Detector
TCS	Omega 10 Whr Heater
TCS	In-house thermal board
AVI	Tyvak Intrepid
AVI	In-house CDH Breakout board

Software	Processor
Image Processing	CFC-300
Star Tracker	CFC-300
Star Subtraction	CFC-300
Object Tracking	CFC-300
ADCS Controller	BBB
ADCS Hardware polling	BBB
Health Monitoring	Tyvak Intrepid
EPS Hardware Controller	Tyvak Intrepid
TCS Hardware Controller	Tyvak Intrepid
Innoflight CFC-300 Hardware Controller	Tyvak Intrepid
BBB Hardware Controller	Tyvak Intrepid
Finite State Machine (FSM)	Tyvak Intrepid
Scheduler	Tyvak Intrepid
File Manager	Tyvak Intrepid
Message Bus	Tyvak Intrepid
Event Log	Tyvak Intrepid
UHF COM Receive	Tyvak Intrepid
UHF COM Transmit	Tyvak Intrepid
GlobalStar COM Receive	Tyvak Intrepid
GlobalStar COM Transmit	Tyvak Intrepid

2   Overview of Flight Software

In the interest of clarifying the nomenclature of the RECONSO AVI team from those used by other groups that are employing the cFE architecture as well, the purpose of each block of FSW will be briefly explained. Please note that this is not being done from a computer science perspective, but rather from a systems-engineering perspective, with the interest of being easy to understand by those without a background in computer science. the reader should refer back to this list throughout the document to aid in the understand of how each piece of software will help accomplish the goal of each mode of the satellite.

1. Image Processing - When analyzing pictures taken by the payload, this block of code will handle the opening and reading of each image file.

2. Star Tracker - While taking pictures, the star tracker will compute angular distance between stars in each picture and compare them to an on-board star catalog to precisely determine the attitude of the spacecraft. This code will run in real time with a frequency of 10 Hz.

3. Star Subtraction - While analyzing pictures to track objects, the star subtraction code will determine which features of the image are stars and remove them from subsequent frames so that they are not tracked as objects.

4. Object Tracking - After the stars in a given image have been subtracted, the object tracking code will track objects in subsequent frames to determine their right ascension and declination relative to the spacecraft as well as their photometric brightness.

5. ADCS Controller - The ADCS controller will analyze measurements taken from each ADCS sensor, take into account each sensor's known uncertainty, and filter this data to determine what action should be taken by the actuators to reach a desired attitude.

6. ADCS Hardware Polling - ADCS hardware polling regards the use of serial communications protocol to request data from each ADCS sensor at a frequency of 10 Hz.

7. Health Monitoring - To ensure that the spacecraft is operating nominally, the health monitoring software will poll various sensors in various subsystems of the spacecraft and compare it to safe values.

8. EPS Hardware Controller - The EPS hardware controller will be used to communicate with the EPS stack to determine which components are receiving power at a given time and how much power they are receiving.

9. TCS Hardware Controller - The TCS hardware controller will poll the thermal board's resistive temperature detectors (RTDs) and heaters to determine the thermal state of the spacecraft and whether or not heaters or components need to be turned on or off.

10. Innoflight CFC-300 Hardware Controller - The Innoflight controller will be used to handle all serial communication when polling payload data from the Innoflight CFC-300. All object tracks and stars in the field of view will be stored locally on the CFC-300 and will be sent to the ADCS Controller as it is requested.

11. BBB Hardware Controller - The BBB controller will handle serial communication between the primary flight computer and the BBB. While the ADCS controller makes attitude determinations, the hardware controller will allow pointing commands and desired attitude of the spacecraft to be passed to the BBB.

12. Finite State Machine - The FSM will receive health data and determine the correct mode of operations for the satellite at any given time. Should a mode switch be necessary, the FSM will determine what pieces of hardware need to be turned on or off and what software processes need to be started or stopped.

13. Scheduler - The scheduler will control when various software apps run so that processes are not started or stopped in the incorrect order. It can be considered to be a timetable of events through which the satellite moves in chronological order.

14. File Manager - When a process needs to retrieve the contents of a memory address on board the spacecraft, the file manager will return those contents to the process that requests them. When a process needs to store a file, the file manager will write the contents of that file to memory and return their memory address.

15. Message Bus - The message bus serves as the communication path along which every process can pass messages to every other process running onboard.

16. Event Log - The event log will be used to recording software events as they happen along with a timestamp. This will be used for the ground to debug any software errors and to ensure nominal function of the satellite.

17. UHF COM Receive - The UHF COM Receive app communicates with the UHF Daughterboard to receive and pass along the message bus any messages that the satellite receives from the ground over UHF.

18. UHF COM Transmit - The UHF COM Transmit app communicates with the UHF Daughterboard to package messages that the satellite sends to the ground over UHF.

19. GlobalStar COM Receive - The GlobalStar COM Receive app serves the same purpose as the UHF COM Receive app, except it communicates with the GlobalStar Duplex.

20. GlobalStar COM Transmit -The GlobalStar COM Transmit app serves the same purpose as the UHF COM Transmit app, except it communicates with the GlobalStar Duplex.

3 Spacecraft Modes of Operation

As was mentioned in the previous section, the mode switching on RECONSO will be handled by the FSM built-in cFE application. The FSM application will command various components and processes to either begin or end operations based on the mode into which it determines the satellite currently needs to operate.

The FSM will make these decisions based off of health data, commands in the scheduler, and the location of the satellite in its orbit. At a basic level, the decisions that the FSM will step through are portrayed in the two images below.

Figure 1: Operation of RECONSO over the course of the entire mission lifetime.
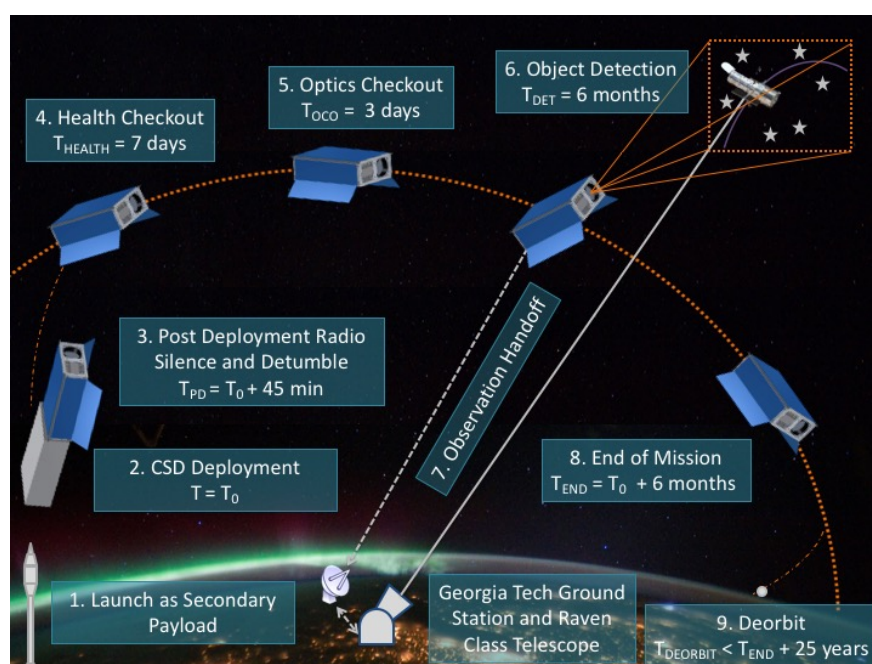
Figure 2: Operation of RECONSO over the course of a given orbit.
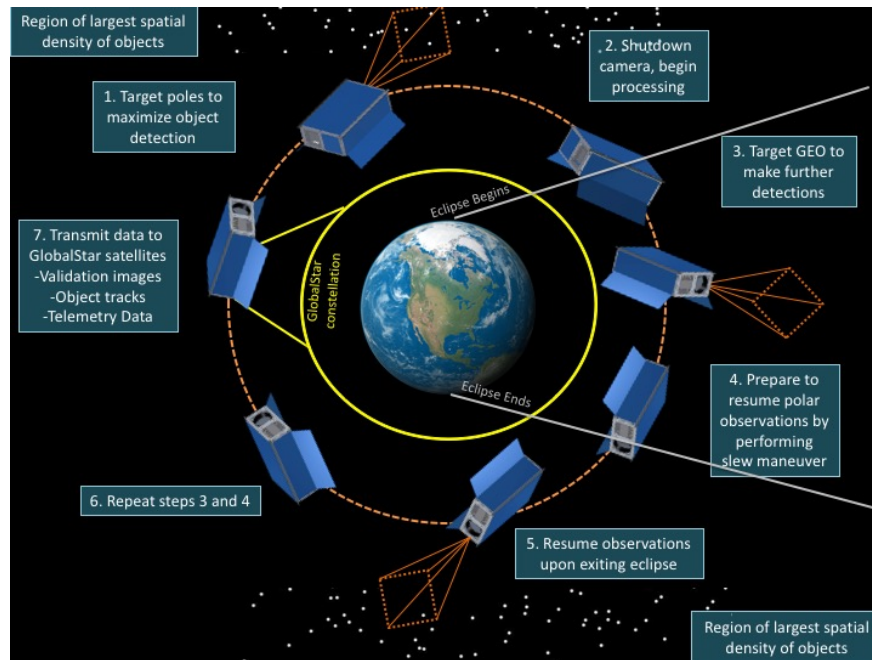
In order to accomplish these operations, the satellite will have 13 modes of operation between which it will switch. The remainder of this document will be dedicated to addressing each of these modes in detail. The overview, entry criteria, exit criteria, operational hardware, and operational software of each mode will be described, along with any additional pertinent information that goes with each mode.

Before beginning, it should be noted that there are certain pieces of hardware and software that will always be used, throughout all modes of spacecraft operation. In the interest of avoiding repetitiveness, these items will be outlined in the table below. They will not be included in the tables defining the operational hardware and software in subsequent mode definitions.

Table 2: Operational components of the RECONSO system throughout all modes.

Subsystem	Hardware
EPS	ClydeSpace 30Whr Battery
EPS	ClydeSpace PMAD Distribution board
EPS	In-house EPS Breakout board
EPS	In-house Inhibit board
EPS	In-house Solar panels
AVI	Tyvak Intrepid
AVI	In-house CDH Breakout board

Software	Processor
Health Monitoring	Tyvak Intrepid
EPS Hardware Controller	Tyvak Intrepid
Finite State Machine (FSM)	Tyvak Intrepid
Scheduler	Tyvak Intrepid
File Manager	Tyvak Intrepid
Message Bus	Tyvak Intrepid
Event Log	Tyvak Intrepid
UHF COM Receive	Tyvak Intrepid
GlobalStar COM Receive	Tyvak Intrepid

It should be noted that the only items operational throughout all modes is only that hardware necessary to the integrity of the spacecraft. The EPS subsystem will always be conducting operations that help determine the health and basic functionality of the satellite. At it's most basic state, the batteries will be used to determine that an unacceptable depth of discharge has been reached and only the bare minimum hardware and software will be run until the batteries can be replenished through power output from the solar panels.

Each of the following subsections will list only hardware and software items that will be used **in addition** to these items.

3.1 Post Deployment

<u>Overview:</u> There are no extra hardware or software items required of the satellite in this mode. The Post Deployment mode of operation will only be entered once throughout the mission lifetime. After being ejected from the CSD as a secondary payload, RECONSO will power on, but only with the bare minimum operation and no radio communications. This is a launch provider requirement so that the communications of secondary payloads do not interfere with communications between the ground and the primary payload or launch vehicle.

<u>Entry Criteria:</u>

1. Confirmation of launch vehicle ejection.

<u>Exit Criteria:</u>

1. Passage of 45 minutes after launch vehicle ejection.

Table 3: Additional components of RECONSO required for Post Deployment mode.

Subsystem	Hardware		Software	Processor
None	None		None	None

3.2   Detumble

<u>Overview:</u> The detumble mode will be used to null any unacceptably high angular rates and to return the spacecraft to a steady attitude. It will primarily be used directly after exiting the post deployment mode, when angular values are likely to be highest due to angular rates imparted by launch vehicle separation. It will also be used at any time throughout the mission when the Health Monitoring software determines that unacceptably high angular rates have been incurred. These rates could be imparted by anything from solar pressure to micrometeoroid impact. Once the angular rates have been nulled, the spacecraft will be placed into safe mode.

<u>Entry Criteria:</u>

1. Exit of Post Deployment mode

2. Unacceptably high angular rates

<u>Exit Criteria:</u>

1. Acceptably low angular rates

Table 4: Additional components of RECONSO required for Detumble mode.

Subsystem	Hardware
ADCS	Analog Devices Inertial Measurement Unit
ADCS	In-house sun sensors
ADCS	In-house magnetorquers
ADCS	Tyvak Magnetometer
ADCS	BBB ADCS Controller

Software	Processor
ADCS Controller	BBB
ADCS Hardware polling	BBB
BBB Hardware Controller	Tyvak Intrepid

3.3   Safe Mode

Overview: Safe mode is designed to be the bare minimum operations that the spacecraft can carry out. It will be placed into this mode upon a reboot of the primary flight computer and at any time throughout spacecraft operations that a health measurement is recorded outside of nominal bounds for a given component, as specified in Appendix B. If the health monitoring code detects unacceptable health values, it will kill any active processes and record in the Event Log the reasoning for entering safe mode. Telemetry will be polled and recorded throughout safe mode operation so that it can be determined when the system is ready to exit safe mode. COM will be enabled during this mode so that the ground can understand why the spacecraft has entered this mode and any changes to the planned operations can be made. Only UHF COM will be utilized in safe mode in order to maximize the power efficiency and protect the battery from critical levels. Fine pointing will not be required as the mostly omnidirectional nature of the UHF COM systems will allow for low-throughput communications that will be required for the ground to accurately diagnose the situation and command any basic operation that will bring the spacecraft out of safe mode. The ADCS and TCS subsystems remain active throughout safe mode to ensure the payload lens remains healthy and does not point directly at the sun or exceed a nominal temperature range.

It should also be noted that every other mode of operation has exit criteria equivalent to the entry criteria of safe mode, i.e. if any of the anomalies that trigger safe mode occur in any other operating mode, that mode is terminated and safe mode is initiated.

Entry Criteria:

1. Temperatures higher than 50 °C

2. Temperatures lower than -10 °C

3. Unacceptable power draw

4. Detection of more than 80% depth of discharge

5. Detection of voltage spike

Exit Criteria:

1. Nominal health readings

2. Ground station receipt of all telemetry leading up to triggering of safe mode

3. Ground station receipt of healthy telemetry after occurrence of anomaly

4. Ground station confirmation that satellite is capable of returning to nominal operation

Table 5: Additional components of RECONSO required for Safe mode.

Subsystem	Hardware
ADCS	Analog Devices Inertial Measurement Unit
ADCS	Pumpkin GPSRM 1 GPS Receiver
ADCS	AntCom 1.5G L1 GPS Antenna
ADCS	In-house sun sensors
ADCS	In-house magnetorquers
ADCS	Tyvak Magnetometer
ADCS	BBB ADCS Controller
COM	Tyvak UHF Daughterboard
COM	ISIS UHF Dipole Antenna
TCS	Omega Resistive Temperature Detector
TCS	Omega 10 Whr Heater
TCS	In-house thermal board

Software	Processor
ADCS Controller	BBB
ADCS Hardware polling	BBB
TCS Hardware Controller	Tyvak Intrepid
BBB Hardware Controller	Tyvak Intrepid
UHF COM Receive	Tyvak Intrepid
UHF COM Transmit	Tyvak Intrepid
GlobalStar COM Receive	Tyvak Intrepid
GlobalStar COM Transmit	Tyvak Intrepid

3.4   Systems Checkouts

Overview: The Systems Checkouts mode will be used for diagnosing the long term health of the spacecraft. As such, it varies from Safe mode only slightly. Systems Checkouts will be used at two different times throughout the lifecycle of the spacecraft: upon exiting safe mode for the first time after launch and upon ground station commanding of this mode. Upon first exiting safe mode, the spacecraft will remain in this mode for 7 days. This will be done to determine that there has not been any subtle degradation of the spacecraft hardware cause by the launch environment and that it is able to function for extended periods of time. During this mode, telemetry will be sent from the spacecraft down to the ground station every hour in the form of a beacon. To ensure the health of GlobalStar system, this beacon will first be sent over GlobalStar COM. However, in order to ensure that the UHF COM is also functional, the ground station will command the satellite to switch to UHF after verifying health of the GlobalStar system. This will allow for proper testing of both COM systems and allow the mission operations team to address any problems with the spacecraft before they become untenable. The spacecraft will exit this mode of operation when commanded to do so by the ground station.

   Entry Criteria:

1. First exit of safe mode

2. Commanding by ground station

   Exit Criteria:

1. Commanding by ground station

Table 6: Additional components of RECONSO required for Systems Checkouts mode.

Subsystem	Hardware
ADCS	Analog Devices Inertial Measurement Unit
ADCS	Pumpkin GPSRM 1 GPS Receiver
ADCS	AntCom 1.5G L1 GPS Antenna
ADCS	In-house sun sensors
ADCS	In-house magnetorquers
ADCS	Tyvak Magnetometer
ADCS	BBB ADCS Controller
COM	ISIS UHF Dipole Antenna
COM	NSL GlobalStar Duplex
COM	NSL GlobalStar Patch Antenna
TCS	Omega Resistive Temperature Detector
TCS	Omega 10 Whr Heater
TCS	In-house thermal board

Software	Processor
ADCS Controller	BBB
ADCS Hardware polling	BBB
TCS Hardware Controller	Tyvak Intrepid
BBB Hardware Controller	Tyvak Intrepid
UHF COM Receive	Tyvak Intrepid
UHF COM Transmit	Tyvak Intrepid
GlobalStar COM Receive	Tyvak Intrepid
GlobalStar COM Transmit	Tyvak Intrepid

3.5 Optics Checkouts

Overview: Similar to Systems Checkouts, Optics Checkouts will be entered immediately upon the first exiting of Systems Checkouts mode or upon commanding by the ground station. This mode is designed to test the integrity of the mission payload by taking a series of images and downlinking them to the ground. The ground will command mission times and pointing commands at which the spacecraft should take an image, and the spacecraft will do so. This will allow further integrated testing of on-orbit performance necessary for nominal operations and will again allow the spacecraft operators to understand any foibles or errors in the spacecraft functionality before nominal operations commence. A minimum of 1 validation image will be downlinked, but more may be requested should the first be deemed insufficient to make a confident decision. Star tracking will also be demonstrated in this mode and the resultant telemetry will be downlinked along with the validation image.

Entry Criteria:

1. First exit of Systems Checkouts Mode

2. Commanding by ground station

Exit Criteria:

1. Commanding by ground station

Table 7: Additional components of RECONSO required for Optics Checkouts mode.

Subsystem	Hardware
PAY	Nocturn 50mm f/0.95 Lens
PAY	Nocturn XL CMOS Camera
PAY	Innoflight CFC-300 Processor
ADCS	Analog Devices Inertial Measurement Unit
ADCS	Pumpkin GPSRM 1 GPS Receiver
ADCS	AntCom 1.5G L1 GPS Antenna
ADCS	In-house sun sensors
ADCS	In-house magnetorquers
ADCS	Tyvak Magnetometer
ADCS	BBB ADCS Controller
COM	ISIS UHF Dipole Antenna
COM	NSL GlobalStar Duplex
COM	NSL GlobalStar Patch Antenna
TCS	Omega Resistive Temperature Detector
TCS	Omega 10 Whr Heater
TCS	In-house thermal board

Software	Processor
Image Processing	CFC-300
Star Tracker	CFC-300
ADCS Controller	BBB
ADCS Hardware polling	BBB
TCS Hardware Controller	Tyvak Intrepid
Innoflight CFC-300 Hardware Controller	Tyvak Intrepid
BBB Hardware Controller	Tyvak Intrepid
UHF COM Receive	Tyvak Intrepid
UHF COM Transmit	Tyvak Intrepid
GlobalStar COM Receive	Tyvak Intrepid
GlobalStar COM Transmit	Tyvak Intrepid

3.6   Sunlight Operations

Overview: The Sunlight Operations mode will be the only mode in which the spacecraft gathers relevant mission data. This mode will be entered once the spacecraft has arrived at a location in its orbit where it needs to gather data (either polar region or approach to the equator), is pointing at the correct attitude to gather data, has determined that it has sufficient disk space to write payload images, and is completely healthy. The spacecraft will then take images at a frequency of 10 Hz until it has passed through the region in which it needed to gather data. While it is taking these images, star tracking will be employed to determine the attitude of the spacecraft. Each image will also be time-tagged so that later analysis can correlate tracks to GPS locations and attitude states.

Entry Criteria:

1. Arrival in region of payload operation

2. Correct execution of pointing command

3. Sufficient disk space for payload operation

4. Nominal health

Exit Criteria:

1. Exit from region of payload operation

Table 8: Additional components of RECONSO required for Sunlight Operations mode.

Subsystem	Hardware
PAY	Nocturn 50mm f/0.95 Lens
PAY	Nocturn XL CMOS Camera
PAY	Innoflight CFC-300 Processor
ADCS	Analog Devices Inertial Measurement Unit
ADCS	Pumpkin GPSRM 1 GPS Receiver
ADCS	AntCom 1.5G L1 GPS Antenna
ADCS	In-house sun sensors
ADCS	Tyvak Magnetometer
ADCS	BBB ADCS Controller
COM	ISIS UHF Dipole Antenna
COM	NSL GlobalStar Duplex
COM	NSL GlobalStar Patch Antenna
TCS	Omega Resistive Temperature Detector
TCS	Omega 10 Whr Heater
TCS	In-house thermal board

Software	Processor
Image Processing	CFC-300
Star Tracker	CFC-300
Star Subtraction	CFC-300
Object Tracking	CFC-300
ADCS Controller	BBB
ADCS Hardware polling	BBB
TCS Hardware Controller	Tyvak Intrepid
Innoflight CFC-300 Hardware Controller	Tyvak Intrepid
BBB Hardware Controller	Tyvak Intrepid
UHF COM Receive	Tyvak Intrepid
GlobalStar COM Receive	Tyvak Intrepid

3.7   Sunlight Data Processing

Overview: If, in evaluating the entry conditions for entrance of Sunlight Operations mode, there is insufficient disk space on the CFC-300 to write images, the spacecraft will instead enter Sunlight Data Processing mode. In this case, the spacecraft will process the images remaining in memory and create a data product, rather than take new images. After completing this process, the spacecraft will enter Sunlight Nominal mode.

Entry Criteria:

1. Arrival in region of payload operation

2. Correct execution of pointing command

3. **In**sufficient disk space for payload operation

4. Nominal health

Exit Criteria:

1. Sufficient disk space for payload operation

Table 9: Additional components of RECONSO required for Sunlight Data Processing mode.

Subsystem	Hardware
ADCS	Analog Devices Inertial Measurement Unit
ADCS	Pumpkin GPSRM 1 GPS Receiver
ADCS	AntCom 1.5G L1 GPS Antenna
ADCS	In-house sun sensors
ADCS	Tyvak Magnetometer
ADCS	BBB ADCS Controller
COM	ISIS UHF Dipole Antenna
COM	NSL GlobalStar Duplex
COM	NSL GlobalStar Patch Antenna
TCS	Omega Resistive Temperature Detector
TCS	Omega 10 Whr Heater
TCS	In-house thermal board

Software	Processor
Image Processing	CFC-300
Star Tracker	CFC-300
Star Subtraction	CFC-300
Object Tracking	CFC-300
ADCS Controller	BBB
ADCS Hardware polling	BBB
TCS Hardware Controller	Tyvak Intrepid
Innoflight CFC-300 Hardware Controller	Tyvak Intrepid
BBB Hardware Controller	Tyvak Intrepid
UHF COM Receive	Tyvak Intrepid
GlobalStar COM Receive	Tyvak Intrepid

3.8 Sunlight Transmit

<u>Overview:</u> Given the quasi-global coverage supplied by the GlobalStar network, the transmission of data from the spacecraft to the ground can occur at any point in the spacecraft's orbit. As such, the spacecraft will, at a minimum enter this mode to downlink health data to the ground. If the spacecraft has made one or more tracks since last entering transmit mode, those tracks will be downlinked as well. Should the spacecraft not register any tracks after data processing, only health data will be downlinked and the FSM will direct the spacecraft straight to Sunlight Nominal Operations. Given that the spacecraft will constantly be able to receive messages while outside of Post Deployment mode, this mode will also be entered should the ground command a response from the spacecraft.

<u>Entry Criteria:</u>

1. Completion of Sunlight Data Processing

2. Commanding by ground station

<u>Exit Criteria:</u>

1. Completion of transmission

Table 10: Additional components of RECONSO required for Sunlight Transmit mode.

Subsystem	Hardware
PAY	Innoflight CFC-300 Processor
ADCS	Analog Devices Inertial Measurement Unit
ADCS	Pumpkin GPSRM 1 GPS Receiver
ADCS	AntCom 1.5G L1 GPS Antenna
ADCS	In-house sun sensors
ADCS	Tyvak Magnetometer
ADCS	BBB ADCS Controller
COM	ISIS UHF Dipole Antenna
COM	NSL GlobalStar Duplex
COM	NSL GlobalStar Patch Antenna
TCS	Omega Resistive Temperature Detector
TCS	Omega 10 Whr Heater
TCS	In-house thermal board

Software	Processor
ADCS Controller	BBB
ADCS Hardware polling	BBB
TCS Hardware Controller	Tyvak Intrepid
Innoflight CFC-300 Hardware Controller	Tyvak Intrepid
BBB Hardware Controller	Tyvak Intrepid
UHF COM Receive	Tyvak Intrepid
UHF COM Transmit	Tyvak Intrepid
GlobalStar COM Receive	Tyvak Intrepid
GlobalStar COM Transmit	Tyvak Intrepid

3.9   Sunlight Nominal

Overview: The Sunlight Nominal mode will be entered once the Sunlight Transmit mode is completed, but while the satellite is still in sunlight and is generating power. During this time, the spacecraft will make any attitude adjustments necessary for the next phase of operations (either a polar or equatorial region) and then wait until it is passes into a region suitable for those operations.

Entry Criteria:

1. Exit from Sunlight Transmit or Sunlight Data Processing modes

Exit Criteria:

1. Accomplishment of next pointing command

2. Entrance into eclipse

Table 11: Additional components of RECONSO required for Sunlight Nominal mode.

Subsystem	Hardware
ADCS	Analog Devices Inertial Measurement Unit
ADCS	Pumpkin GPSRM 1 GPS Receiver
ADCS	AntCom 1.5G L1 GPS Antenna
ADCS	In-house sun sensors
ADCS	Tyvak Magnetometer
ADCS	In-house magnetorquers
ADCS	BBB ADCS Controller
COM	ISIS UHF Dipole Antenna
COM	NSL GlobalStar Duplex
COM	NSL GlobalStar Patch Antenna
TCS	Omega Resistive Temperature Detector
TCS	Omega 10 Whr Heater
TCS	In-house thermal board

Software	Processor
ADCS Controller	BBB
ADCS Hardware polling	BBB
TCS Hardware Controller	Tyvak Intrepid
Innoflight CFC-300 Hardware Controller	Tyvak Intrepid
BBB Hardware Controller	Tyvak Intrepid
UHF COM Receive	Tyvak Intrepid
GlobalStar COM Receive	Tyvak Intrepid

3.10   Eclipse Data Processing

Overview: The Eclipse Data Processing mode is very similar to Sunlight Data Processing in that it processes whatever is left of the payload image data after it has been gathered and saved to disk. However, there is a difference in hardware used during this mode as the spacecraft should be in a lower power state while in eclipse so as not reach an unacceptable depth of discharge. In addition, due to the heavy power requirement of the Innoflight CFC-300 hardware, this state must be entered by command from the ground station upon verification that there is a sufficient battery charge remaining.

It is assumed that the spacecraft will not be collecting data while it is in eclipse, so if there is no data to be processed, this mode will not be entered.

Entry Criteria:

1. Entrance into eclipse

2. Unprocessed payload data still on disk

3. Command from ground station after verification of sufficient battery charge

Exit Criteria:

1. Exit of eclipse or finishing of data processing, whichever comes first.

Table 12: Additional components of RECONSO required for Eclipse Data Processing mode.

Subsystem	Hardware
PAY	Innoflight CFC-300 Processor
ADCS	In-house sun sensors
ADCS	Tyvak Magnetometer
ADCS	BBB ADCS Controller
COM	ISIS UHF Dipole Antenna
COM	NSL GlobalStar Duplex
COM	NSL GlobalStar Patch Antenna
TCS	Omega Resistive Temperature Detector
TCS	Omega 10 Whr Heater
TCS	In-house thermal board

Software	Processor
Image Processing	CFC-300
Star Subtraction	CFC-300
Object Tracking	CFC-300
ADCS Controller	BBB
ADCS Hardware polling	BBB
TCS Hardware Controller	Tyvak Intrepid
Innoflight CFC-300 Hardware Controller	Tyvak Intrepid
BBB Hardware Controller	Tyvak Intrepid
UHF COM Receive	Tyvak Intrepid
GlobalStar COM Receive	Tyvak Intrepid

3.11 Eclipse Transmit

Overview: Much like Sunlight Transmit mode, Eclipse Transmit will occur if, once data processing has completed, there is payload data that has not been downlinked to the ground. However, it differs from Sunlight Transmit in that, it will not downlink telemetry unless commanded by the ground station to do so. Due to the high power requirements of transmission, it is desirable to limit transmission time while in eclipse as much as is possible. Once all necessary data has been transmitted, RECONSO will exit this mode.

Entry Criteria:

1. Completion of Eclipse Data Processing

2. Payload data on disk that has not been downlinked

3. Command from ground station after verification of sufficient battery charge

Exit Criteria:

1. Completion of transmission or lack thereof

Table 13: Additional components of RECONSO required for Eclipse Transmit mode.

Subsystem	Hardware
PAY	Innoflight CFC-300 Processor
ADCS	In-house sun sensors
ADCS	Tyvak Magnetometer
ADCS	BBB ADCS Controller
COM	ISIS UHF Dipole Antenna
COM	NSL GlobalStar Duplex
COM	NSL GlobalStar Patch Antenna
TCS	Omega Resistive Temperature Detector
TCS	Omega 10 Whr Heater
TCS	In-house thermal board

Software	Processor
Image Processing	CFC-300
Star Subtraction	CFC-300
Object Tracking	CFC-300
ADCS Controller	BBB
ADCS Hardware polling	BBB
TCS Hardware Controller	Tyvak Intrepid
Innoflight CFC-300 Hardware Controller	Tyvak Intrepid
BBB Hardware Controller	Tyvak Intrepid
UHF COM Receive	Tyvak Intrepid
UHF COM Transmit	Tyvak Intrepid
GlobalStar COM Receive	Tyvak Intrepid
GlobalStar COM Transmit	Tyvak Intrepid

3.12 Eclipse Nominal

Overview: Upon completion of Eclipse Transmit mode or Eclipse Data Processing mode (or having never entered either), the spacecraft will enter Eclipse Nominal mode. Upon verification of a sufficient battery charge, the next pointing command relevant to the gathering of payload data will be executed. Given that the spacecraft is in eclipse, this will be towards one of the poles as opposed to towards the equator. Upon completing this slew maneuver, the spacecraft will wait until entering sunlight and further payload operations.

If the battery charge level is deemed insufficient to avoid entering safe mode due to depth of discharge, this state will not execute a pointing command and therefore idle until the end of eclipse.

Entry Criteria:

1. Completion of Eclipse Transmit

Exit Criteria:

1. Accomplishment of next pointing command

2. Entrance into sunlight

3. Pointing command executed upon verification of sufficient battery power remaining

Table 14: Additional components of RECONSO required for Eclipse Nominal mode.

Subsystem	Hardware
ADCS	In-house sun sensors
ADCS	Tyvak Magnetometer
ADCS	BBB ADCS Controller
COM	ISIS UHF Dipole Antenna
COM	NSL GlobalStar Duplex
COM	NSL GlobalStar Patch Antenna
TCS	Omega Resistive Temperature Detector
TCS	Omega 10 Whr Heater
TCS	In-house thermal board

Software	Processor
ADCS Controller	BBB
ADCS Hardware polling	BBB
TCS Hardware Controller	Tyvak Intrepid
BBB Hardware Controller	Tyvak Intrepid
UHF COM Receive	Tyvak Intrepid
GlobalStar COM Receive	Tyvak Intrepid

3.13 End of Life

Overview: After a mission lifetime of 6 months or upon incurring a fatal hardware error, the spacecraft will be put into End of Life mode. The goal of the End of Life mode is to ensure that the spacecraft does not transmit or receive on any frequencies while it safely de-orbits within the 25 year limit. Given that RECONSO is employing solely passive de-orbit procedures, there is a possibility that this mode will last for quite a long time. The ground station will command a series of commands the spacecraft that will allow it to shut down without the aid of the ground station. The commands will be as follows: put the satellite into safe mode, command the PMAD to shunt all power generated by the solar panels, turn off the flight computer. This will eventually drain the batteries without leaving them a possibility or recharging, and RECONSO will die a natural death.

It should be noted that while COM hardware and software is listed as being necessary for this mode, it is only necessary for the entrance to this mode. It will not be operation while within End of Life mode. Should all COM hardware be anomalous, the end goal of this mode will have already been accomplished and thus will not be necessary.

Entry Criteria:

1. Commanding by ground station

Exit Criteria:

1. None

Table 15: Additional components of RECONSO required for End of Life mode.

Subsystem	Hardware
COM	ISIS UHF Dipole Antenna
COM	NSL GlobalStar Duplex
COM	NSL GlobalStar Patch Antenna

Software	Processor
UHF COM Receive	Tyvak Intrepid
GlobalStar COM Receive	Tyvak Intrepid

4 Conclusion

There are a total of 13 modes of operation of RECONSO, each accomplishing a unique function. To reiterate, the FSM code will handle all switching between each mode autonomously based on pre-defined entry an exit criteria. Once the nominal mission lifetime of RECONSO has been accomplished, the End of Life mode will be commanded, and the mission will be permanently ended. Please reference Appendix A for a visual flow chart representation of this same information.

5   Appendix A:

Commands and Nominal Telemetry Values

ADCS Commands	Nominal Result
ADCS_Telemetry GET_ADCS_TEL()	See specific telemetry points below
float* MAGTORQ_CUR_CMD()	-0.3 to 0.3
float* V_ECI()	-1000 to 1000
float* Q_ECI_TO_BODY()	-7.67 to 7.67
float* ANGVEL_BODY()	-1 to 1
float VAR_ECI_POS()	-250 to 250 tumbling, -10 to 10 nominal
float VAR_ECI_VEL()	TBD
float VAR_BORESIGHT_ATTITUDE()	TBD
uint8_t POINTINGMODE()	1, 2, 3 ,4
float Q_ERROR_NORM()	0 to 30
uint8_t SUNSENSOR_FLAG()	0 or 1
uint8_t MAGMETER_FLAG()	0 or 1
uint8_t FILTER_RESET_FLAG()	0 or 1
uint16_t FILTER_RESET_TIMESTAMP()	Time
float* GYRO_GYROFRAME_MEAS()	-250 to 250 tumbling, -10 to 10 nominal
float* MAGMETER_MAGMETERFRAME_MEAS()	TBD
float* ACCELMETER_ACCELMETERFRAME_MEAS()	TBD
float* STARTRACKER_ECI_TO_BODY_MEAS()	0 to 1
double GPS_DATA()	TBD
float* SUNSENSOR_MEAS()	TBD
TORQUE_ROD_X_CURRENT(float xcurr)	-0.3 to 0.3
TORQUE_ROD_Y_CURRENT(float ycurr)	-0.3 to 0.3
TORQUE_ROD_Z_CURRENT(float zcurr)	-0.3 to 0.3
ADCS_Ack UPDATE_CONTROLLER_GAINS()	TBD
ADCS_Ack UPLOAD_POINTING_PROFILE(uint8_t pprof)	0 or 1
ADCS_Ack RESET_FILTER(uint8_t reset)	0 or 1
ADCS_Ack SET_POINTING_MODE(uint8_t pmode)	0, 1, 2, . . . , 5, 6
ADCS_Ack SET_AD_SENSORS(uint8_t adsensors)	0, 1, 2, . . . , 9, 10

CDH Commands	Nominal Result
CDH_Telemetry GET_CDH_TEL()	See specific telemetry points below
uint32_t TYV_CPU_USERTIME(void)	time
uint32_t TYV_CPU_NICETIME(void)	time
uint32_t TYV_SYS_TIME(void)	time
uint32_t TYV_IDLE_TIME(void)	time
uint32_t TYV_VIRT_PG_IN(void)	0 to TBD
uint32_t TYV_VIRT_PG_OUT(void)	0 to TBD
uint32_t TYV_PG_SWPS_IN_MEM(void)	0 to TBD
uint32_t TYV_PG_SWPS_OUT_MEM(void)	0 to TBD
uint32_t TYV_NUM_INTERUPTS(void)	0 to TBD
uint32_t TYV_NUM_PROC_CONTEXT_SWPS(void)	0 to TBD
uint32_t TYV_TIME_FROM_BOOT(void)	time
uint32_t TYV_TOTAL_PROCESSES(void)	0 to TBD
uint32_t TYV_RUN_PROCESSES(void)	0 to TBD
uint32_t TYV_BLOCKED_PROCESSES(void)	0 to TBD
uint32_t TYV_FREE_MEM(void)	0 to 512 MB
uint32_t TYV_BUFF_MEM(void)	0 to 512 MB
uint32_t TYV_CACHED_MEM(void)	0 to 512 MB
uint32_t TYV_ACTIVE_MEM(void)	0 to 512 MB
uint32_t TYV_VIRT_MEM_TOTAL(void)	0 to TBD
uint32_t TYV_VRT_MEM_USED(void)	0 to TBD
uint32_t TYV_MEM_FLASH_FREE(void)	0 to 12 GB
uint32_t TYV_MEM_SD_FREE(void)	0 to 32 GB TBR
uint32_t TYV_TIME_LINUX_EPOCH(void)	time
uint16_t TYV_CNTR_LONG_DURATION_STATUS(void)	time

COM Commands	Nominal Result
COMM_Telemetry GET_COMM_TEL()	See specific telemetry points below
uint8_t DIPOLE_DEPLOYMENT()	0 or 1
uint8_t DIPOLE_STATUS()	0 or 1
float DIPOLE_TEMPERATURE()	-20 to 100
uint32_t DUPLEX_RESET()	0 to TBD
uint32_t DUPLEX_SECONDS()	time
uint8_t DUPLEX_RSSI()	-100 to 0
uint8_t DUPLEX_GATEWAY()	TBD
uint32_t DUPLEX_LAST_CONTACT()	time
uint32_t DUPLEX_LAST_ATTEMPT()	time
uint32_t DUPLEX_NUMBER_CALL()	0 to TBD
uint32_t DUPLEX_NUMBER_SUCCESS()	0 to TBD
float DUPLEX_AVG_CONNECT()	time
float DUPLEX_STDDEV_CONNECT()	0 to TBD

EPS Commands	Nominal Result
EPS_Telemetry GET_EPS_TEL()	See specific telemetry points below
int16_t BAT_BOARD_STATUS()	8 bit response IAW Battery manual Table 12-2
int16_t BAT_LAST_ERROR()	16 bit response IAW Battery manual Table 12-3
uint16_t BAT_VERSION()	16 bit response IAW Battery manual Table 12-4
int16_t BAT_CHECKSUM()	16 bit response IAW Battery manual Table 12-5
int16_t BAT_TELEM()	8 bit response IAW Battery manual Section 12.5
uint16_t BAT_BROWNOUT_RESETS()	0 to 255
uint16_t BAT_AUTO_RESETS()	0 to 255
uint16_t BAT_GET_MANUAL_RESETS()	0 to 255
int16_t BAT_GET_HEATER_CONTROL_STAT()	16 bit response IAW Battery manual Table 12-8
BAT_SET_HEATER_CONTROL_STAT()	EPS_Ack
BAT_SET_MANUAL_RESET()	EPS_Ack
CS_PCM_RESET()	EPS_Ack
int32_t CS_VERSION()	4 byte response IAW CS manual table on page 46
CS_SET_SYSTEM_WATCHDOG_TIMEOUT()	EPS_Ack
CS_RESET_SYSTEM_WATCHDOG()	EPS_Ack
uint16_t CS_NUM_SYSTEM_RESETS()	0 to 255
CS_PDM_INITIAL_STATE_ON()	EPS_Ack
CS_PDM_ALL_PDM_ON()	EPS_Ack
CS_PDM_ALL_PDM_OFF()	EPS_Ack
uint16_t CS_PDM_ACTUAL_STATUS()	0 to 1
uint16_t CS_PDM_INIT_STATE()	0 to 1
uint16_t CS_ANALG_TELEM(uint8_t device)	0 to 1023
uint16_t CS_GET_SYSTEM_WATCHDOG_TIMEOUT()	1 to 90
CS_PDM_SWITCH_ON()	EPS_Ack
CS_PDM_SWITCH_OFF()	EPS_Ack
uint16_t CS_GET_NUM_SOFT_RESETS()	0 to 255
int16_t CS_EXPECTED_PDM_SWITCH_STATE()	10 bit response IAW CS manual Table 11
uint16_t CS_BOARD_TEMP_COUNT()	0 to 1024
CS_RESET_NODE()	EPS_Ack

PAY Commands	Nominal Result
PAY_Telemetry GET_PAY_TEL()	See specific telemetry points below
uint32_t MIN_THRESHOLD;	TBD
uint32_t MAX_THRESHOLD;	TBD
uint32_t THRESHOLD_STEP;	TBD
uint32_t MIN_BLOB_AREA;	TBD
uint32_t MAX_BLOB_AREA;	TBD
uint32_t NUM_DETECTIONS;	0 to TBD
double MEAN_SIZE;	TBD
uint32_t ICP_ITERATION_LIMIT;	TBD
double ICP_CONVERGENCE_THRESHOLD;	TBD
uint32_t ICP_MAX_POINTS_CONSIDERED;	TBD
PAY_Image GET_RAW_IMAGE()	image data
PAY_ProcessedImage GET_PROCESSED_IMAGE()	processed image points
PAY_StarTracker GET_STAR_TRACKS()	star tracker points

TCS Commands	Nominal Result
TCS_Telemetry GET_TCS_TEL()	See specific telemetry points below
uint16_t TEMP_RTD()	244 to 311 K nominal, 73.15 to 787.15 degree K total range
uint8_t STATUS_HEATER()	0 or 1
SET_HEATER_ON()	1
SET_HEATER_OFF()	0