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clear all; clc; close all;
mu = 3.9860044189e5;
R1 = [3286 5010 9787];
R2 = [-3259 260 13009];
R3 = [-11787 -7674 9628];

r1 = norm(R1);
r2 = norm(R2);
r3 = norm(R3);
N = r1*(cross(R2,R3)) + r2*(cross(R3,R1)) + r3*(cross(R1,R2));
D = cross(R2,R3) + cross(R3,R1) + cross(R1,R2);
S = R1*(r2-r3) + R2*(r3-r1) + R3*(r1-r2);

V2 = sqrt(mu/(dot(N,D)))*(cross(D,R2)/norm(R2)+S);

r = R2;
v = V2;
[h,e,i,w,W,true_ana] = stateVec2OrbElem(r,v);

v2 = sqrt(mu/(dot(N,D)))*(cross(D,R2)/norm(R2)+S);

rp = norm(h)^2/mu*1/(1+norm(e))-6378;
fprintf('True Anamoly: %0.3f degrees \n', true_ana)
fprintf('Specific Angular Momentum: %0.0f km^2/s \n', norm(h))
fprintf('Eccentricity: %0.3f \n', norm(e))
fprintf('Inclination: %0.3f degrees \n', i)
fprintf('Right ascension of the ascending Node: %0.3f degrees \n', W)
fprintf('Argument of perigee %0.3f degrees \n', w)

if i < 90
    fprintf('The orbit is Prograde: i<90 \n')
else
    fprintf('The orbit is Retrograde: i>90 \n')
end
if true_ana > 180
    fprintf('The satellite is approaching perigee \n')
else
    fprintf('The satellite is going away perigee \n')
end
fprintf('Perigee Altitude %0.3f km \n', rp)

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function [RX,VX,r,v,QXx] = OrbElem2StateVec(h,e,i,w,W,true_ano)
%This function will take orbital elements and compute two state vectors r and
v
% r and v must be 3-D vectors
mu = 3.9860044189e5;
r = (h^2/mu)/(1+e*cos(true_ano))*[cos(true_ano) sin(true_ano) 0]';
v = mu/h*[-sin(true_ano) (e+cos(true_ano)) 0]';
R3_W = [cos(W) sin(W) 0;
        -sin(W) cos(W) 0;
        0 0 1];
R1_i = [1 0 0;
        0 cos(i) sin(i);
        0 -sin(i) cos(i)];
R3_w = [ cos(w) sin(w) 0;
        -sin(w) cos(w) 0
        0 0 1];
QXx = (R3_w) *(R1_i) *(R3_W);

RX = QXx.*r;
VX = QXx.*v;

end

```

True Anomaly: 80.000 degrees

Specific Angular Momentum: 75001 km<sup>2</sup>/s

Eccentricity: 0.300

Inclination: 79.999 degrees

Right ascension of the ascending Node: 39.999 degrees

Argument of perigee 20.001 degrees

The orbit is Prograde: i<90

The satellite is going away perigee

Perigee Altitude 4477.849 km