Control Barrier Functions

Internship M1

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Subject

- High Relative Degree Control Barrier Functions Under Input Constraints from Joseph Breeden and Dimitra Panagou (2021)
- Control Barrier Functions (CBF) to avoid collisions
- Understand the article, the case of simply avoiding the obstacle
- New objective: the case of getting closer to a target while avoiding the obstacle
- No time for 2 satellites

Some notation

If $h: \mathbb{R}^n \to \mathbb{R}$ and $f: \mathbb{R}^n \to \mathbb{R}^n$:

$$\mathcal{L}_f h(x) = \langle \nabla h(x), f(x) \rangle$$

where $\nabla h(x)$ is the gradient of h at the point x.

If $h: \mathbb{R}^n \to \mathbb{R}$ and $q: \mathbb{R}^n \to \mathbb{R}^{n \times m}$, we reformulate this derivative as

$$\mathcal{L}_g h(x) = \begin{bmatrix} \langle \nabla h(x), g_1(x) \rangle \\ \vdots \\ \langle \nabla h(x), g_m(x) \rangle \end{bmatrix}$$

where $g(x) = (g_1(x), ..., g_m(x))$ and $g_i : \mathbb{R}^n \to \mathbb{R}^n$.

Formulation of the problem

$$\begin{cases} \dot{y}(t) = \begin{bmatrix} \dot{r}(t) \\ \dot{v}(t) \end{bmatrix} = \begin{bmatrix} v(t) \\ u(t) \end{bmatrix} = \underbrace{\begin{bmatrix} v(t) \\ 0_3 \end{bmatrix}}_{f(y(t))} + \underbrace{\begin{bmatrix} 0_{3 \times 3} \\ I_{3 \times 3} \end{bmatrix}}_{g(y(t))} u(t) \\ y(0) = x \end{cases}$$

 \iff

$$\begin{cases} \dot{y}(t) = f\left(y(t)\right) + g\left(y(t)\right)u\left(y(t)\right) = F\left(t, y(t)\right) \\ y(0) = x \end{cases}$$

 $\text{where} \left\{ \begin{array}{l} x \in \mathbb{R}^n, \\ u \in U \text{ compact}, \\ f : \mathbb{R}^n \to \mathbb{R}^n \in \mathcal{C}^r, \\ g : \mathbb{R}^n \to \mathbb{R}^{n \times m} \in \mathcal{C}^r. \end{array} \right.$

Definitions & Properties

Relative degree 2 functions

 $h:\mathbb{R}^n\to\mathbb{R}$ is said to be of relative degree 2 with respect to the previous dynamics if

- 1. $h \in \mathcal{C}^2$ on an open,
- 2. $\mathcal{L}_g h(x) = 0 \ \forall x \in \mathbb{R}^n$
- 3. $\exists C \subseteq \mathbb{R}^n$ such that $\mathcal{L}_g \mathcal{L}_f h(x) \neq 0 \ \forall x \in C$.

Let \mathcal{G}^2 be the space of functions of relative degree 2 satisfying the given system.

Let $h \in \mathcal{G}^2$, we define the safe set S such that

$$S = \{ x \in \mathbb{R}^n \mid h(x) \le 0 \}.$$

Let

$$H(x) = \sup_{t \ge 0} h\left(\varphi_u\left(t, x\right)\right)$$

where $t\mapsto \varphi_u(t,x)$ is the solution of the previous ODE with $\varphi_u(0,x)=x$ and

$$S_H = \{ x \in \mathbb{R}^n \mid H(x) \le 0 \}.$$

Now we set a control u^* that we have previously defined.

ZCBF

For our dynamics, a function of class \mathcal{C}^1 $h:\mathbb{R}^n\to\mathbb{R}$ is a zeroing control barrier function (ZCBF) on S if

$$\inf_{u \in U} \left[\dot{h}(x, u) + h(x) \right] \le 0 \ \forall x \in S$$

Lemma

If h is a ZBCF on S under our dynamics then for any Lipschitzian controller u such that

$$\dot{h}(x,u) \le -h(x) \ \forall x \in S$$

will make the set S forward invariant.

H is a ZCBF

It can be shown that H is a ZCBF. Thus we have the following implications :

$$\begin{split} H \ \text{ZCBF} &\Rightarrow \exists u \in U \ \dot{H}(x,u) + H(x) \leq 0 \\ &\Rightarrow \text{We can make } S_H \text{ forward invariant.} \end{split}$$

Illustration

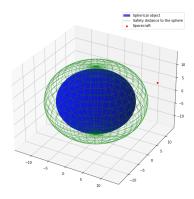


Figure 1: Illustration of the problem

- Spherical object : radius ρ_a and center r_a
- Minimum distance from the sphere : ho_s
- $\bullet\,$ Distance between the satellite and the safety bubble : h

$$h(y(t)) = \rho - \|r_a - r(t)\|_2$$
 where $\rho = \rho_a + \rho_s$ and we take $u^* = u_{ball}.$

Sketch of $H(x) = \sup \varphi_{u_{ball}}(t, x)$

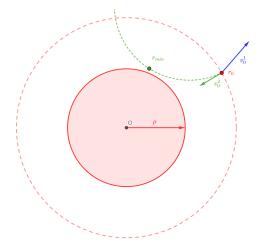


Figure 2: Illustration of H

Control Lyapunov Function

Control Lyapunov Function (CLF)

$$V(y,t) = \frac{1}{2} ||r - r_p(t)||_2^2 + \frac{1}{2} k_2 ||v - k_1(r - r_p(t))||_2^2$$

with k_1 and k_2 as constants.

Optimisation problem

The control function we are looking for here is defined by the following expression:

$$\tilde{u}(x,t) = \underset{(u,\delta) \in C}{\operatorname{argmin}} u^T u + k\delta^2$$
$$= \underset{(u,\delta) \in C}{\operatorname{argmin}} \|(u,\sqrt{k}\delta) - (0_3,0)\|_2^2$$

$$C = \left\{ (u, \delta) \in \mathbb{R}^3 \times \mathbb{R} \mid \begin{array}{c} \mathcal{L}_f H(x) + \mathcal{L}_g H(x) u + H(x) \leq 0 \\ \mathcal{L}_f V(x, t) + \mathcal{L}_g V(x, t) u + \delta k_3 V(x, t) \leq 0 \\ \|u\|_2 - u_{max} \leq 0 \end{array} \right\}.$$

Set of constraints

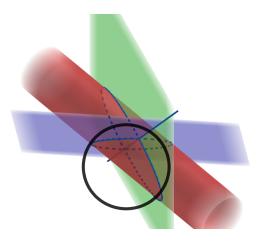


Figure 3: Illustration of constraints with Geogebra 3D

Resolution

We want to solve the following differential equation

$$\begin{cases} \dot{y}(t) = \begin{bmatrix} \dot{r}(t) \\ \dot{v}(t) \end{bmatrix} = \begin{bmatrix} v(t) \\ \tilde{u}(y(t), t) \end{bmatrix} \\ y(0) = x \end{cases}$$

where \tilde{u} is the solution of the following problem

$$\tilde{u}(x,t) = \underset{(u,\delta) \in C}{\operatorname{argmin}} u^T u + k\delta^2$$

How to solve integration

$$\dot{y}(t) = \begin{bmatrix} v(t) \\ \tilde{u}(y(t), t) \end{bmatrix} \implies \mathsf{RK4}\left(k_1, k_2, k_3, k_4\right)$$

Calculate \tilde{u} for each k



Solve optimisation problem



Determine C



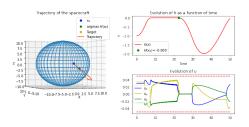
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Calculate H and \dot{H}

Another integration to find the \dot{H}

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Target point and conflict path



Trajectory of the spacecraft

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Figure 4: Target point

Figure 5: Conflict path

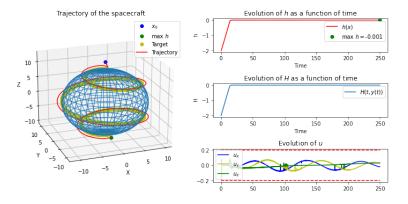


Figure 6: Beautiful trajectory

Gravitation

$$f(y) = \begin{bmatrix} v \\ f_{\mu}(r) \end{bmatrix}$$
 with $f_{\mu}(r) = \mu \frac{r_a - r}{\|r_a - r\|^3}$

where $\mu = \mathcal{G}M$ with G the gravitational constant and M the mass of the object.

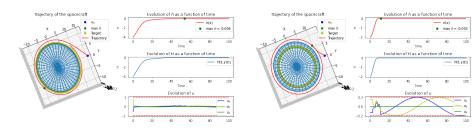
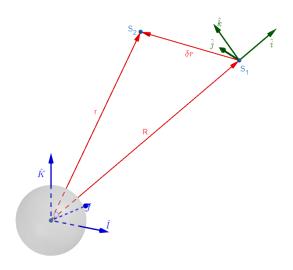


Figure 7: $\mu = 0$

Figure 8: $\mu = 50$

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Illustration of the problem

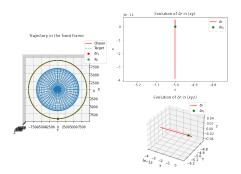


Linearization of the equations of relative motion in orbit

 $\left\{ \begin{array}{l} R,V: \ \ \text{The position and speed of} \ S_1 \\ \delta r,\delta v: \ \ \text{The position and speed of} \ S_2 \ \text{in the motion frame} \\ \Omega = \frac{h}{\|R\|}: \ \ \text{The angular acceleration of} \ S_1 \end{array} \right.$

$$\begin{cases} \dot{R} = V \\ \dot{V} = -\mu \frac{R}{\|R\|^3} \\ \delta \dot{r} = \delta v \\ \delta \ddot{x} = \left(\frac{2\mu}{\|R\|^3} + \frac{\|h\|^2}{\|R\|^4} \right) \delta x - 2\|h\| \frac{\langle V \mid R \rangle}{\|R\|^4} \delta y + 2 \frac{\|h\|}{\|R\|^2} \delta \dot{y} \\ \delta \ddot{y} = -\left(\frac{\mu}{\|R\|^3} - \frac{\|h\|^2}{\|R\|^4} \right) \delta y + 2\|h\| \frac{\langle V \mid R \rangle}{\|R\|^4} \delta x - 2 \frac{\|h\|}{\|R\|^2} \delta \dot{x} \\ \delta \ddot{z} = -\frac{\mu}{\|R\|^3} \delta z \end{cases}$$

Trajectories



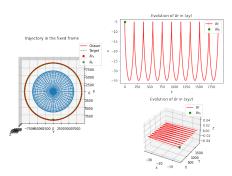


Figure 10: Circular orbit and $\delta r_0 = (0, -5, 0)$ **Figure 11:** Circular orbit and $\delta r_0 = (-5, 0, 0)$

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Trajectories

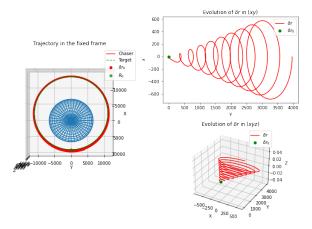


Figure 12: Elliptical orbit and $\delta r_0 = (-5, 0, 0)$

Hill-Clohessy-Wiltshire equations

$$\begin{cases} \langle V \mid R \rangle = 0, \\ \|h\| = \sqrt{\mu \|R\|} \end{cases}$$

$$\delta r(t) = \Phi_{rr}(t)\delta r_0 + \Phi_{rv}(t)\delta v_0$$

and

$$\delta v(t) = \Phi_{vr}(t)\delta r_0 + \Phi_{vv}(t)\delta v_0$$

where

$$\Phi_{rr}(t) = \begin{bmatrix} 4 - 3\cos(nt) & 0 & 0\\ 6(\sin(nt) - nt) & 1 & 0\\ 0 & 0 & \cos(nt) \end{bmatrix}$$

. . .

Trajectories

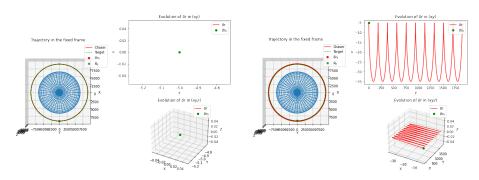


Figure 13: $\delta r_0 = (0, -5, 0)$

Figure 14: $\delta r_0 = (-5, 0, 0)$

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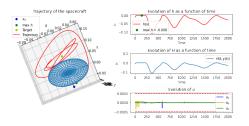
$$f_n(\delta r, \delta v) = \begin{bmatrix} 3n^2 \delta x + 2n\delta \dot{y} \\ -2n\delta \dot{x} \\ -n^2 \delta z \end{bmatrix}.$$

$$f(\delta r, \delta v) = \begin{bmatrix} \delta v \\ f_n(\delta r, \delta v) \end{bmatrix}$$

$$h(y) = \rho - \|\delta r\|$$

 $u^* = u_{ball}$.

Trajectories



Trajectory of the spacecraft

| Second | Second

Figure 15: With ${\cal H}$

Figure 16: Without ${\cal H}$

New optimisation problem

$$\tilde{u}(x,t) = \underset{\|u\| \le u_{max}}{\operatorname{argmin}} \mathcal{L}_f V + \mathcal{L}_g V u$$

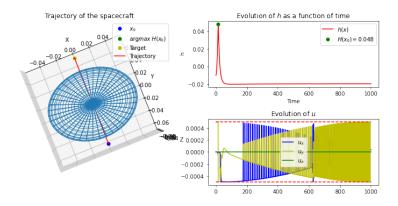
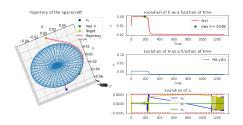


Figure 17: New optimisation problem for the attraction

$$\tilde{u}(x,t) = \begin{cases} \underset{\|u\| \le u_{max}}{\operatorname{argmin}} \mathcal{L}_f V + \mathcal{L}_g V u & \text{if } H(x) \le -m \\ \underset{\|u,\delta) \in C}{\operatorname{argmin}} u^T u + k \delta^2 & \text{else} \end{cases}$$



Trajectory of the spacecraft Evolution of h as a function of time max h = -0.000 Evolution of H as a function of time - H(t, y(t)) Evolution of u 400 600 800 1000

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Figure 18: m = 0.002

Figure 19: m = 0.008

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