Reaction Control System for High Altitude Balloons

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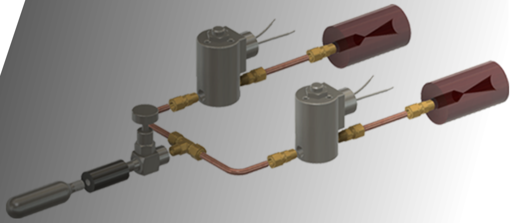


Nozzles

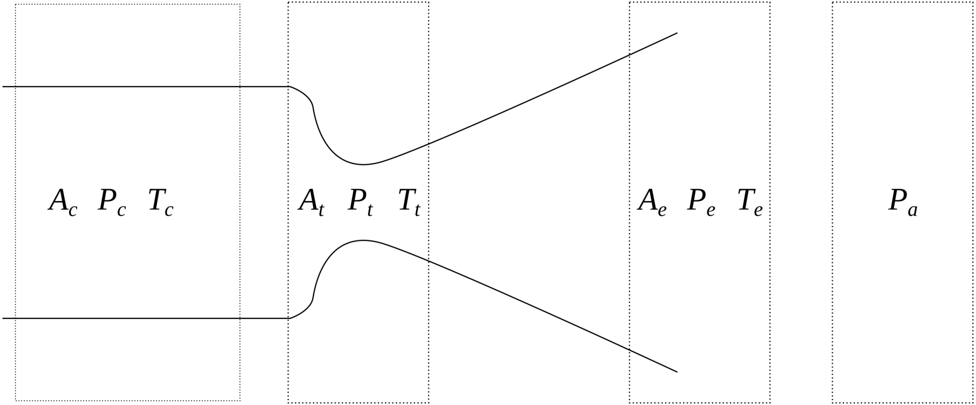


Figure: Basic nozzle design and some parameters.

Existing Nozzle Theory

$$\frac{A_e}{A_t} = \left(\frac{\gamma+1}{2}\right)^{\frac{1}{1-\gamma}} \left(\frac{P_a}{P_c}\right)^{\frac{1}{\gamma}} \left(\left(\frac{\gamma+1}{\gamma-1}\right)\left[1-\left(\frac{P_c}{P_a}\right)^{\frac{\gamma-1}{\gamma}}\right]\right)^{-\frac{1}{2}} \tag{1}$$

Optimum expansion ratio for nozzle

Nozzle Design

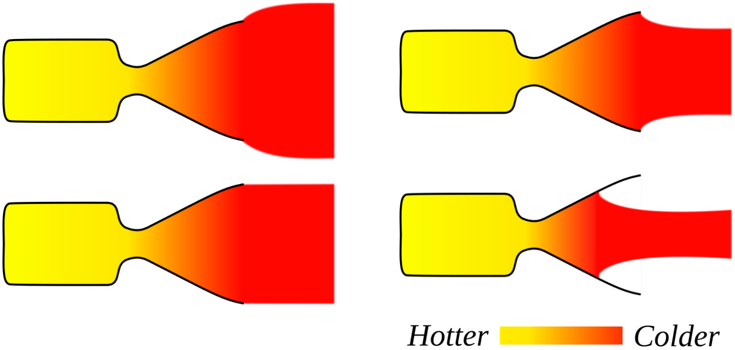


Figure: Nozzle expansion and temperature gradient visualized.

Specific Impulse

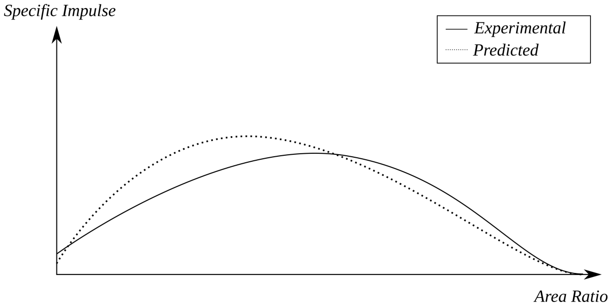
Definition of specific impulse:

$$I_{sp} = \frac{\textit{Thrust}}{\textit{rate of mass ejection}} \tag{2}$$

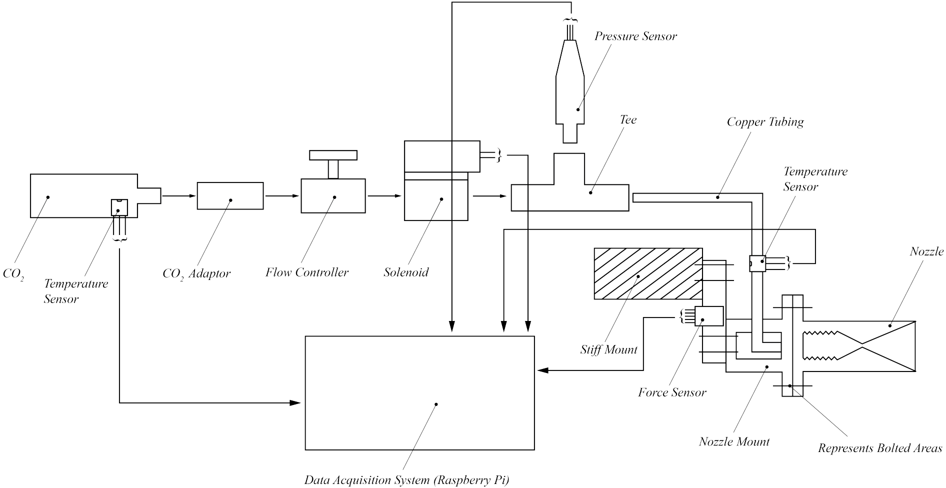
$$I_{sp} = \left(\frac{2\gamma G T_c}{(\gamma - 1)W} \left(1 - \left(\frac{T_e}{T_c}\right)\right)^{\frac{1}{2}} \right) + \frac{A_e}{A_t} \left(\left(\frac{T_e}{T_c}\right)^{\frac{\gamma}{\gamma-1}} \right) \sqrt{\frac{P_c}{\gamma \rho_c}} \left(\frac{\gamma + 1}{2}\right)^{\frac{\gamma+1}{2(\gamma-1)}} \tag{3}$$

Derived specific impulse

Develop an Experiment



Develop an Experiment



Manufacturing

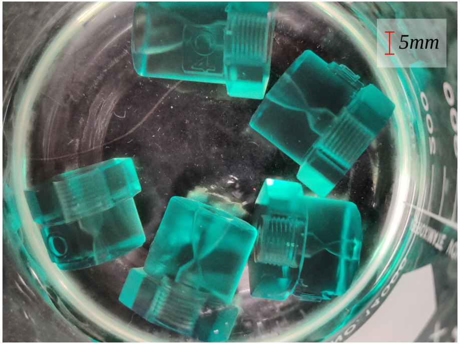
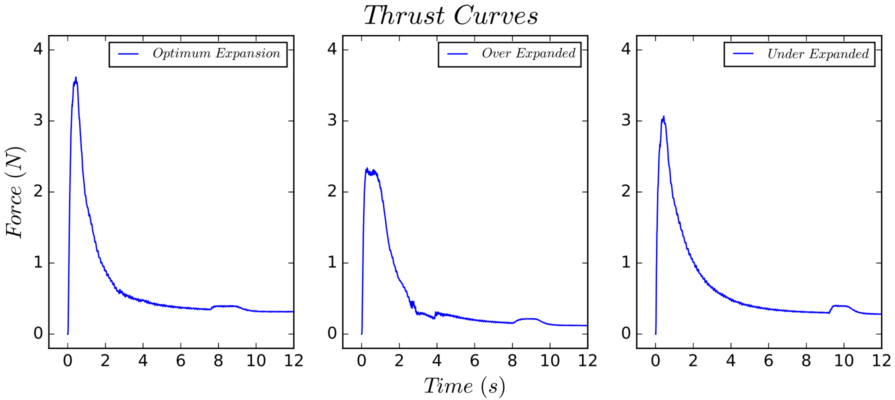


Figure: Nozzles bathing in isopropyl alcohol.

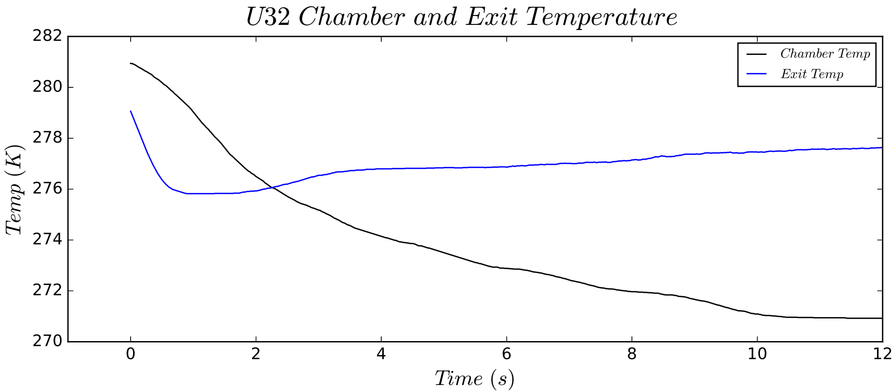
Data Collection



Analysis

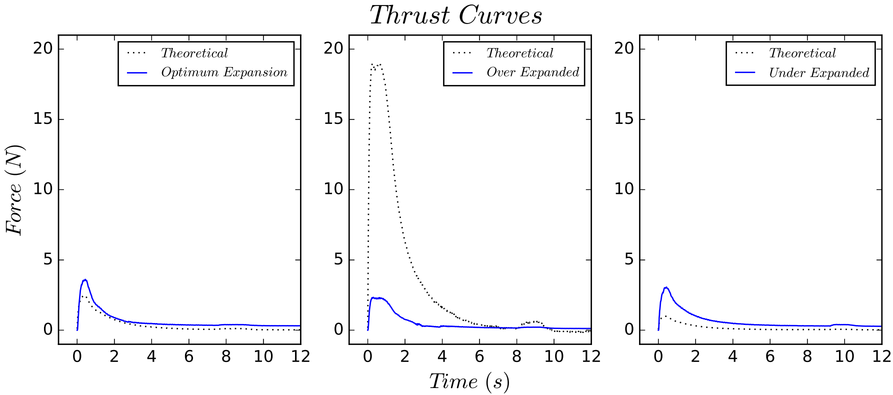
$$F = A_t P_c \left(\sqrt{\frac{2\gamma^2}{\gamma - 1} \left(\frac{2}{\gamma + 1} \right)^{\frac{\gamma+1}{\gamma-1}} \left(1 - \frac{T_e}{T_c} \right)} + \left(\left(\frac{T_e}{T_c} \right)^{\frac{\gamma}{\gamma-1}} - \frac{P_a}{P_c} \right) \frac{A_e}{A_t} \right) \tag{4}$$

Analysis



U32 Chamber and Exit Temperature
Chamber Temp
Exit Temp
Temp (K)
282
280
278
276
274
272
270
0
2
4
6
8
10
12
Time (s)

Analysis


Thrust Curves
Force (N)
Time (s)
Theoretical
Optimum Expansion
Theoretical
Over Expanded
Theoretical
Under Expanded
0
5
10
15
20
0
2
4
6
8
10
12

Payload Integration

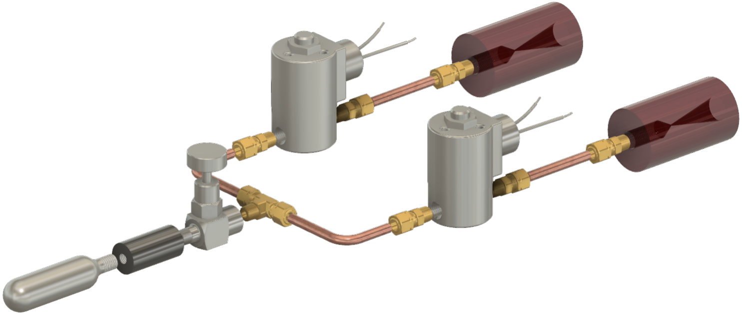


Figure: CAD Render of the plumbing system for my RCS.

Problems and Solutions

1. Exit plane temperature
2. Mass flow rate
3. Constant pressure for best geometry
4. Data acquisition solutions

Temperature Data

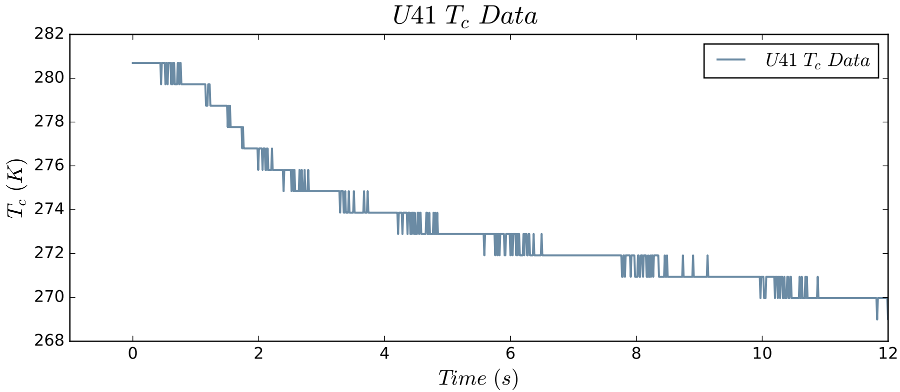


Figure: Temperature plot showing the ADC’s lack of accuracy

Conclusion

1. Missing some information to complete characterization of the CGT
2. Solid foundation for future work
3. Integration into HABP
4. Flight on board a HAB

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