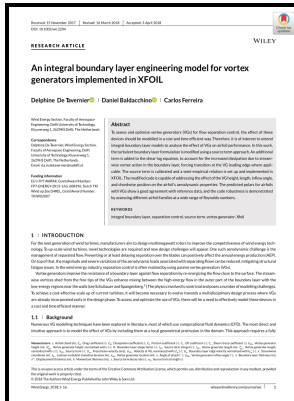


# Use of hot-film technique for boundary layer studies on a 21% thick airfoil

National Aeronautical Establishment - The use of hot



Description: -

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Transonic speed  
Hot-film anemometers

Boundary layer flow

Airfoil use of hot-film technique for boundary layer studies on a 21% thick airfoil

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The higher turbulence in the initial region of the jet shear layer leads to increased high-frequency noise. Each method has advantages and limitations.

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The procedure was developed for calculating three dimensional, incompressible turbulent boundary layers was expanded to adiabatic, compressible flows. The conditions and apparent reasons for laminar boundary layers which have been observed during previous investigations on the walls of several nozzles for exit Mach numbers from 2 to 20 are reviewed. The MSES code is used for a prediction.

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In the present paper the magnetohydrodynamic boundary-layer equation is also expressed in a simple form that is especially convenient for physical interpretation. The inlet was flush mounted to the tunnel wall and ingested a large boundary layer with a boundary-layer-to-inlet height ratio of 35%.

DTIC ADA184412: The Use of Hot

The previously reported unusual result NACA Technical Note 4037 that a highly cooled stagnation point flow is more unstable than a highly cooled flat-plate flow is again encountered.

The Use of Hot

Reversing flow near the wall is the first sign of incipient separation and can bristle shark scales which may be linked to a passive, flow actuated separation control mechanism.

## **The Use of Hot**

The experimental data are taken from an experiment of W. The acetone is atomized by the high shear at the wall into a 'fog' of tiny  $\sim 3$ .

## **laminar boundary**

For a wide angle of attack range below stall, the flow separates laminar from the suction surface. It was shown that two-dimensional geometries like a groove are more stable than three-dimensional geometries like a dimple. The data show that the flow control jets alone can reduce circumferential distortion  $DPC_{avg}$  from 0.

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