

## **Model identification and flight control design for the Prometheus mapping drone**

**Candidate:** Nicola Dal Lago

**Advisor:** Prof. Luca Schenato

**Advisor:** Prof. George Nikolakopoulos

**Advisor:** Emil?

Master in Control Engineering  
Department of Information Engineering  
2016



## Abstract

Write here the abstract



# Contents

<b>Contents</b>	<b>v</b>
<b>1 Introduction</b>	<b>1</b>
<b>2 Design and model</b>	<b>5</b>
2.1 Mechanical design . . . . .	5
2.2 Mathematical model . . . . .	6
2.2.1 Quaternion math . . . . .	7
2.2.2 Quadrotor modelling . . . . .	10
2.2.3 Adding the rotating platform . . . . .	12
2.3 Experimental setup . . . . .	14
<b>3 System identification</b>	<b>17</b>
3.1 System simplification and linear approximation . . . . .	17
3.2 Quadrotor parameters . . . . .	19
3.3 Kalman filter . . . . .	19
3.4 Results . . . . .	21
<b>4 Trajectories generator</b>	<b>25</b>
4.1 Trajectory definition . . . . .	25
4.2 Optimization of a trajectory between two setpoints . . . . .	27
4.3 Optimization of a trajectory between $m + 1$ setpoints . . . . .	29
4.4 Adding corridor constraints . . . . .	33
<b>5 Control</b>	<b>35</b>
5.1 Position tracking controller in $SO(3)$ . . . . .	35
5.1.1 Adding the rotating platform . . . . .	37
5.1.2 Simulation Results . . . . .	38
<b>6 Conclusions and future works</b>	<b>41</b>
<b>Bibliography</b>	<b>43</b>



# 1

## Introduction

In these last years, a growing interest has been shown in robotics. In fact, several industries (automotive, medical, manufacturing, space, etc.), require robots to replace men in dangerous, repetitive or onerous situations. A wide area of this research is dedicated to Unmanned Aerial Vehicle (UAV) and especially the one of having the capability of Vertical TakeOff and Landing (VTOL) [1]. This kind of vehicle can be used in a variety of different scenarios, for its reasonable price, small dimensions and great sensors capability. In particular, nowadays intensive research has been accomplished in the area of environment monitoring and exploration, performing with different strategies and sensors.



Figure 1.1: An example of UAV. T-Hawk, a US-made UAV, commonly used to search for roadside bombs in Iraq, made its debut when it photographed the Fukushima nuclear plant from above, providing a detailed look at the interior damage.

Many types of UAVs have been developed over the last years, in particular the quadrotor type [2]. The aim of this thesis is to contribute to the development of the so-called *Prometheus mapping drone*, a fully autonomous vertical takeoff and landing vehicle, able to perform indoor environment exploration and mapping. To do this, we were inspired from the film *Prometheus*, where drones are able to map an indoor cave. Of course, due to technology and budget limitations, the vehicle will not have the same performance, but will have in theory the same capabilities. As previously said, this thesis is only a part of the project, that has been divided in three main parts:

- mechanical design and building of the UAV [3];
- mathematical model, system identification and control;
- usage of the sensors, mapping and navigation algorithms.

This thesis will focus on the second point, mathematical model, system identification and control, but briefly introductions will give also in the other two points, in particular in the mechanical design, necessary for develop a mathematical model.



Figure 1.2: Frame of the prometheus movie, where the drone is performing the exploration and mapping of the cave.

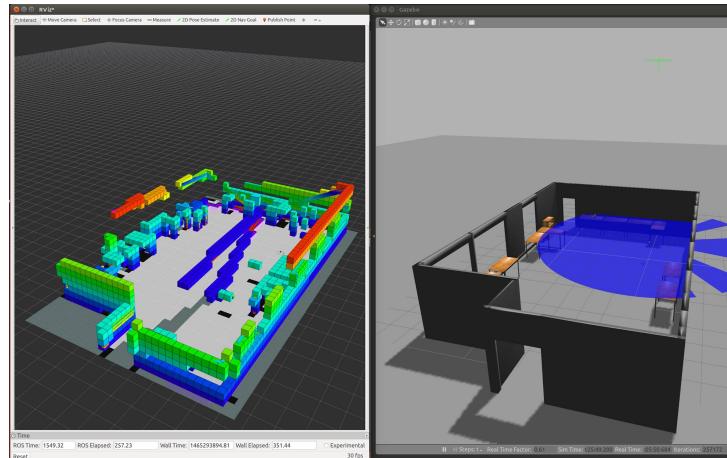


Figure 1.3: Simulation of the navigation and mapping from the third point of this thesis.

In details, this thesis is divided in differnt chapters.

In chapter 2 we will discuss about the mechanical design of the UAV, the sensors used and the reasons for it. From this, a mathematical model will be derive, complete with all the notation and math needed to describe the dynamic of the vehicle. Of course, reasonable simplifications will be perform on the way. Then we will describe the experimental setup, with all the hardware and software used for this project.

In chapter 3 we will perform the system identification of the parameters of the UAV. Before doing it, we will do simplification of the system, just to reduce the number of the parameters and especially to linearize it. The reason to linearize the system, introducing some errors, is because we choose to use a standerd Kalman filter approach to the system identification problem. With this approach we will see that all the parameters will be estimated.

In chapter 4 a trajectories generator algorithm will be describe. This is necessary because usually, path finding and navigation algorithms provides only setpoint and not full trajectory that smoothly connect the different setpoints. We will describe a trajectory generator that smoothly connects setpoint till the fourth derivative of the position, the so called snap. Moreover, corridor constraints will be added to the problem. A corridor constraint takes into account that the trajectory must be inside a virtual corridor between two setpoints. This because if we not impose any constraints we van end up with a trajectory that actually connects different setpoints, but maybe hit a wall. This is extrimly important, since our main goal is to operate the vehicle in a indoor environment.

In chapter 5 we will talk about the control of the vehicle. For control we mean that the UAV must follow the desire trajectory in the space with the minimum error. This control will take into account also the particular structure of the drone. It will be derive from the mathematical model describe previously.

In the end, in chapter 6 we will draw the conclusions of this work. Moreover, we will describe also future possible works, that start from some consideration of this thesis.



# 2

## Design and model

In this chapter we will focus in the description of the mechanical model of the UAV and the sensor system and, from these, a mathematical model will be derived, necessary for build and simulate a control law, and to perform system identification.

### 2.1 Mechanical design

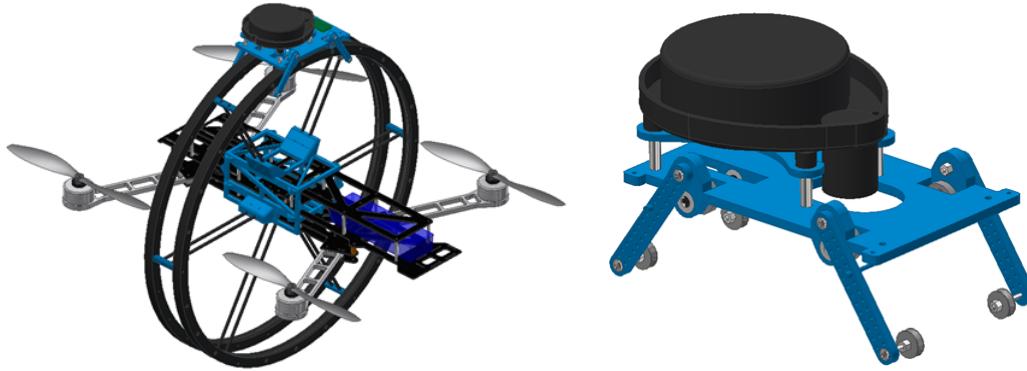
The overall objective of the Prometheus project is navigation and mapping, for these we mean obtain a 3D reconstruction of an unknown indoor physical environment, using a 360 degrees *Lidar* laser scanner, which, coupled to a quadrotor type UAV, will explore in a autonomous way. Lidar is a surveying technology that measures distance by illuminating a target with a laser light. Lidar is an acronym of Light Detection And Ranging, (sometimes Light Imaging, Detection, And Ranging).



Figure 2.1: Lidar laser scanner, it is able to perform a 360 degrees mapping.

Lidar is popularly used as a technology to make high-resolution maps, with appli-

cations in geodesy, geomatics, archaeology, geography, geology, geomorphology, seismology, forestry, atmospheric physics and so on. What is known as Lidar is sometimes simply referred to as laser scanning or 3D scanning, with terrestrial, airborne and mobile applications<sup>1</sup>. The specific Lidar laser scanner used in this project is depicted in figure 2.1, where it is possible to see the rotating structure moved by a motor attached in the bottom of the frame. However, this sensor is only able to perform 2D mapping and, attached to a drone, make it practically impossible to perform a complete 3D mapping. To solve this problem, several approaches could be adopted, such as use a more complicated and more expensive sensor, that can map directly in 3D, or just by simply use more than one Lidar. However, the solution adopted in this project is again inspired from the movie Prometheus, where the sensor is also rotating around the UAV. In such a way, the Lidar has three degrees of freedom in the movement and a 3D mapping can be performed. This solution complies, of course, the usage of only one laser scanner, but requires a rotating structure that can move the sensor.



(a) Render of the entire vehicle.

(b) Render of the moving cart.

Figure 2.2: Renders of the UAV and of the cart.

In figure 2.2a is possible to see clearly the platform, made of two lightweight rings, and the cart that provides the circular movement of the sensor. An important choice was also the selection of the UAV, that has to guarantee to fly also with the weight of the mechanical structure, sensor and all the electronics needed to fly and control the movement of the cart.

## 2.2 Mathematical model

It is pretty clear from the previous section that this UAV is different from almost every other vehicle that is possible to buy, this of course requires a complete and detailed study to characterize the mathematical model. For characterizing the model, it is necessary to provide some definitions, that are also valid for standard commercial quadrotors.

A quadrotor helicopter is made of a central frame and four propellers that are attached to the frame with respectively four arms. Moreover, the propellers' rotation direction must be opposite in pairs, like illustrated in figure 2.3.

Furthermore, it is necessary to define two frames, the world fixed frame and the body frame attached to the vehicle.

<sup>1</sup><https://en.wikipedia.org/wiki/Lidar>

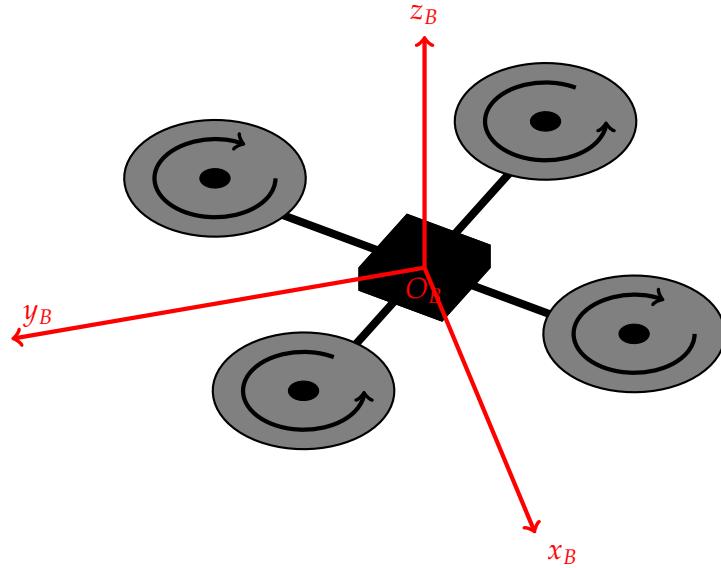


Figure 2.3: Sketch of a standard quadrotor with its body frame attach.

In figure 2.4 is possible to see the two frames, the world frame, in black, is fixed to a point and it can't be moved, the body frame, in blue, instead is attached to the quadrotor and can move with six degrees of freedom, that are position and orientation. In this, we are interesting in knowing the traslation and rotation of the body frame in respect to the world frame. For rappresent the traslation, a three dimension vector  $\mathbf{x}$  is enough, that actually indicate the position of the quadrotor in the space. Instead, for the rotation, we used quaternions [4], that will be introduced in the following section.

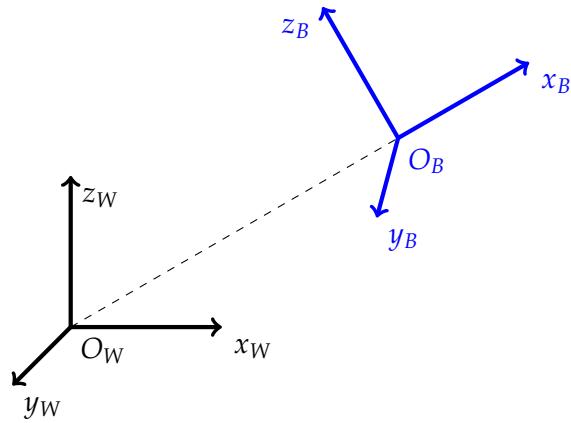


Figure 2.4: Illustration of the world and body frames.

### 2.2.1 Quaternion math

A quaternion is a hyper complex number of rank 4, which can be represented as follow

$$\mathbf{q} = [q_0 \quad q_1 \quad q_2 \quad q_3]^T \quad (2.1)$$

The quaternion units from  $q_1$  to  $q_3$  are called the vector part of the quaternion, while  $q_0$  is the scalar part [5]. Multiplication of two quaternions  $\mathbf{p}$  and  $\mathbf{q}$ , is being performed by the Kronecker product, denoted as  $\otimes$ . If  $\mathbf{p}$  represents one rotation and  $\mathbf{q}$  represents another rotation, then  $\mathbf{p} \otimes \mathbf{q}$  represents the combined rotation.

$$\mathbf{p} \otimes \mathbf{q} = \begin{bmatrix} p_0q_0 - p_1q_1 - p_2q_2 - p_3q_3 \\ p_0q_1 + p_1q_0 + p_2q_3 - p_3q_2 \\ p_0q_2 - p_1q_3 + p_2q_0 + p_3q_1 \\ p_0q_3 + p_1q_2 - p_2q_1 + p_3q_0 \end{bmatrix} \quad (2.2)$$

$$= Q(\mathbf{p})\mathbf{q} = \begin{bmatrix} p_0 & -p_1 & -p_2 & -p_3 \\ p_1 & p_0 & -p_3 & p_2 \\ p_2 & p_3 & p_0 & -p_1 \\ p_3 & -p_2 & p_1 & p_0 \end{bmatrix} \begin{bmatrix} q_0 \\ q_1 \\ q_2 \\ q_3 \end{bmatrix} \quad (2.3)$$

$$= \bar{Q}(\mathbf{q})\mathbf{p} = \begin{bmatrix} q_0 & -q_1 & -q_2 & -q_3 \\ q_1 & q_0 & q_3 & -q_2 \\ q_2 & -q_3 & q_0 & q_1 \\ q_3 & q_2 & -q_1 & q_0 \end{bmatrix} \begin{bmatrix} p_0 \\ p_1 \\ p_2 \\ p_3 \end{bmatrix} \quad (2.4)$$

The norm of a quaternion is defined as

$$\|\mathbf{q}\| = \sqrt{q_0^2 + q_1^2 + q_2^2 + q_3^2} \quad (2.5)$$

If the norm of the quaternion is equal to 1, then the quaternion is called unit quaternion. The complex conjugate of a quaternion has the same definition as normal complex numbers.

$$\mathbf{q}^* = [q_0 \ -q_1 \ -q_2 \ -q_3]^T \quad (2.6)$$

The inverse of a quaternion is defined as a normal inverse of a complex number.

$$\mathbf{q}^{-1} = \frac{\mathbf{q}^*}{\|\mathbf{q}\|^2} \quad (2.7)$$

The time derivative of the unit quaternion is the vector of quaternion rates [6]. It requires some algebraic manipulation but is important to notice that the quaternion rates,  $\dot{\mathbf{q}}$ , are related to the angular velocity  $\boldsymbol{\omega} = [\omega_x \ \omega_y \ \omega_z]^T$ . It can be represented in two ways:

- as in equation (2.8) in case that the angular velocity is in the world frame

$$\dot{\mathbf{q}}_w(\mathbf{q}, \boldsymbol{\omega}) = \frac{1}{2}\mathbf{q} \otimes \begin{bmatrix} 0 \\ \boldsymbol{\omega} \end{bmatrix} = \frac{1}{2}Q(\mathbf{q}) \begin{bmatrix} 0 \\ \boldsymbol{\omega} \end{bmatrix} \quad (2.8)$$

- as in equation (2.9) if the angular velocity vector is in the body frame of reference.

$$\dot{\mathbf{q}}_{w'}(\mathbf{q}, \boldsymbol{\omega}') = \frac{1}{2} \begin{bmatrix} 0 \\ \boldsymbol{\omega}' \end{bmatrix} \otimes \mathbf{q} = \frac{1}{2} \bar{Q}(\mathbf{q}) \begin{bmatrix} 0 \\ \boldsymbol{\omega}' \end{bmatrix} \quad (2.9)$$

A unit quaternion can be used also as a rotation operator, however the transformation requires both the quaternion and its conjugate, as show in equation (2.10). This rotates the vector  $\mathbf{v}$  from the world frame to the body frame represented by  $\mathbf{q}$ .

$$\omega = \mathbf{q} \otimes \begin{bmatrix} 0 \\ \mathbf{v} \end{bmatrix} \otimes \mathbf{q}^* \quad (2.10)$$

Unit quaternion can be use also to represents rotation matrixes. Consider a vector  $\mathbf{v}$  in the world frame. If  $\mathbf{v}'$  is the same vector in the body coordinates, the the following relations hold

$$\begin{bmatrix} 0 \\ \mathbf{v}' \end{bmatrix} = \mathbf{q} \cdot \begin{bmatrix} 0 \\ \mathbf{v} \end{bmatrix} \cdot \mathbf{q}^* \quad (2.11)$$

$$= \bar{Q}(\mathbf{q})^T Q(\mathbf{q}) \begin{bmatrix} 0 \\ \mathbf{v} \end{bmatrix} \quad (2.12)$$

$$= \begin{bmatrix} 1 & \mathbf{0}^T \\ \mathbf{0} & R_{\mathbf{q}}(\mathbf{q}) \end{bmatrix} \begin{bmatrix} 0 \\ \mathbf{v} \end{bmatrix} \quad (2.13)$$

where

$$R_{\mathbf{q}}(\mathbf{q}) = \begin{bmatrix} q_0^2 + q_1^2 - q_2^2 - q_3^2 & 2q_1q_2 + 2q_0q_3 & 2q_1q_3 - 2q_0q_2 \\ 2q_1q_2 - 2q_0q_3 & q_0^2 - q_1^2 + q_2^2 - q_3^2 & 2q_2q_3 + 2q_0q_1 \\ 2q_1q_3 + 2q_0q_2 & 2q_2q_3 - 2q_0q_1 & q_0^2 - q_1^2 - q_2^2 + q_3^2 \end{bmatrix} \quad (2.14)$$

That is,

$$\mathbf{v}' = R_{\mathbf{q}}(\mathbf{q})\mathbf{v} \quad (2.15)$$

$$\mathbf{v} = R_{\mathbf{q}}(\mathbf{q})^T \mathbf{v}' \quad (2.16)$$

Just as with rotation matrices, sequences of rotations are represented by products of quaternions. That is, for unit quaternions  $\mathbf{q}$  and  $\mathbf{p}$ , it holds that

$$R_{\mathbf{q}}(\mathbf{q} \cdot \mathbf{p}) = R_{\mathbf{q}}(\mathbf{q})R_{\mathbf{q}}(\mathbf{p}) \quad (2.17)$$

Finally, for representing quaternion rotations in a more intuitive manner, the conversion from Euler angles (roll  $\phi$ , pitch  $\theta$  and yaw  $\psi$ ) to quaternion and viceversa can be performed by utilizing the following two equations respectively.

$$q = \begin{bmatrix} \cos(\phi/2)\cos(\theta/2)\cos(\psi/2) + \sin(\phi/2)\sin(\theta/2)\sin(\psi/2) \\ \sin(\phi/2)\cos(\theta/2)\cos(\psi/2) - \cos(\phi/2)\sin(\theta/2)\sin(\psi/2) \\ \cos(\phi/2)\sin(\theta/2)\cos(\psi/2) + \sin(\phi/2)\cos(\theta/2)\sin(\psi/2) \\ \cos(\phi/2)\cos(\theta/2)\sin(\psi/2) - \sin(\phi/2)\sin(\theta/2)\cos(\psi/2) \end{bmatrix} \quad (2.18)$$

$$\begin{bmatrix} \phi \\ \theta \\ \psi \end{bmatrix} = \begin{bmatrix} \text{atan2}(2(q_0q_1 + q_2q_3), q_0^2 - q_1^2 - q_2^2 + q_3^2) \\ \text{asin}(2(q_0q_2 - q_3q_1)) \\ \text{atan2}(2(q_0q_3 + q_1q_2), q_0^2 + q_1^2 - q_2^2 - q_3^2) \end{bmatrix} \quad (2.19)$$

### 2.2.2 Quadrotor modelling

We consider first a standard quadrotor, without a rotating platform, like in figure 2.5.

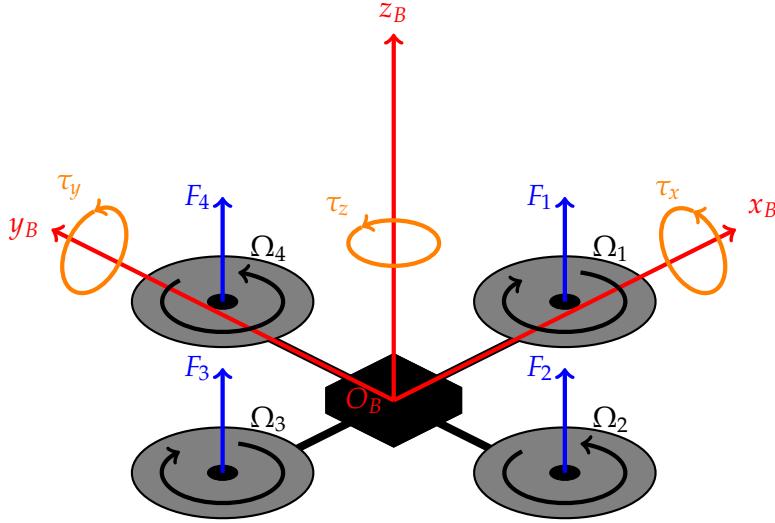


Figure 2.5: Sketch of a standard quadrotor.

In figure 2.5 are also impressed the force vectors  $F_i$  generated from each motor-propeller, the torque vectors  $\tau_x$ ,  $\tau_y$  and  $\tau_z$  about the three axis and the propeller's speed  $\Omega_i$ . Now, for modeling the rigid body of a multirotor, the standard Newton-Euler kinematics equations can be utilized [7].

$$\begin{bmatrix} \mathbf{F} \\ \boldsymbol{\tau} \end{bmatrix} = \begin{bmatrix} m \cdot I_{3 \times 3} & \mathbf{0} \\ \mathbf{0}^T & I_{cm} \end{bmatrix} \begin{bmatrix} \ddot{\mathbf{x}}_B \\ \dot{\boldsymbol{\omega}}_B \end{bmatrix} + \begin{bmatrix} \mathbf{0} \\ \boldsymbol{\omega}_B \times I_{cm} \cdot \boldsymbol{\omega}_B \end{bmatrix} \quad (2.20)$$

Where  $\mathbf{F} = [F_x \ F_y \ F_z]^T$  is the vector of the total force,  $\boldsymbol{\tau} = [\tau_x \ \tau_y \ \tau_z]^T$  is the total torque,  $m$  is the mass of the quadrotor,  $I_{cm}$  is the matrix of inertia related to the center of mass,  $\ddot{\mathbf{x}}_B$  is the acceleration of the quadrotor center of mass related to the body frame and  $\boldsymbol{\omega}_B = [\omega_x \ \omega_y \ \omega_z]^T$  is the rotational rates in the body frame.

Before deriving the torque relationship, the motors' models from the input signal to the thrust force are needed. In specific, the four input signals are the speed of the propellers  $u_i$ , map between 0 (no throttle) and 1 (full throttle). Then, the thrust for each propeller can be simply derived as follows

$$F_i(t) = A_{F,i} \Omega_i^2 = A_{F,i} \Omega_{max,i}^2 u_i(t)^2 \quad (2.21)$$

where  $A_{F,i} \in \mathbb{R}_+$  are the thrust constants of the motor-propeller combination,  $\Omega_{max,i} \in \mathbb{R}_+$  are the maximum rotational speed of the motors and  $u_i(t)$  are the motors' signals. What is missing in equation (2.21) is the model of the DC motors and in particular, a map between the input signal  $u_i(t)$  and the control signal  $u_{in,i}(t)$ . To keep the model simple but still accurate<sup>2</sup>, the motor has been modeled like a delay, like in equation

<sup>2</sup><http://pi19404.github.io/pyVision/2015/04/10/25/>

(2.22).

$$u_i(t) \approx \frac{1}{\tau_i s + 1} u_{in,i}(t) \quad (2.22)$$

This approach is very common [8], since all the parameters of a motor are not provided from datasheet, especially from cheap motors that is possible to find quite often in a commercial quadrotor. Furthermore, to represent the direction of the thrust from a motor it should be considered that

$$\mathbf{F}_i(t) = A_{F,i} \Omega_{max,i}^2 u_i(t) \mathbf{n}_i \quad (2.23)$$

$$\mathbf{n}_i = R_i \cdot [0 \ 0 \ 1]^T \quad (2.24)$$

Where, in this case,  $\mathbf{F}_i(t)$  is the force vector for each propeller and  $R_i$  is the rotational matrix encoding the direction of the thrust and torque vector. Then the torque representation is given by

$$\boldsymbol{\tau}_i(t) = -\text{sgn}(\Omega_i) B_{F,i} \Omega_{max,i}^2 u_i(t)^2 \mathbf{n}_i \quad (2.25)$$

where  $B_{F,i} \in \mathbb{R}_+$  is the torque constant.

Now, by defining the vector  $\mathbf{l}_i = [l_{x,i} \ l_{y,i} \ l_{z,i}]^T$  the distance between the center of mass and the position where the propeller  $i$  is attached, combining equations (2.23), (2.24) and (2.25) is possible to obtain equation (2.26) as in the work [9].

$$\begin{bmatrix} \mathbf{F}_{total} \\ \boldsymbol{\tau}_{total} \end{bmatrix} = \begin{bmatrix} \sum_{i=1}^4 \mathbf{F}_i(u_i^2) \\ \sum_{i=1}^4 \mathbf{l}_i \times \mathbf{F}_i(u_i^2) + \boldsymbol{\tau}_i(u_i^2) \end{bmatrix} \quad (2.26)$$

This combined with the Newton-Euler kinematics of equation (2.20) gives the final model, from control signal to acceleration and angular acceleration, as depicted in equations (2.27) and (2.28).

$$\begin{bmatrix} \ddot{\mathbf{x}}_B \\ \dot{\boldsymbol{\omega}}_B \end{bmatrix} = \begin{bmatrix} \dots & \frac{A_{F,i} \Omega_{max,i}^2 \mathbf{n}_i}{m} \\ \dots & I_{cm}^{-1} \left[ (\mathbf{l}_i + \Delta \mathbf{l}) \times A_{F,i} \Omega_{max,i}^2 \mathbf{n}_i - \text{sgn}(\Omega_i) B_{F,i} \Omega_{max,i}^2 \mathbf{n}_i \right] \end{bmatrix} \begin{bmatrix} \vdots \\ u_i^2 \\ \vdots \end{bmatrix} + \\ + \begin{bmatrix} \mathbf{0} \\ I_{cm}^{-1} (\boldsymbol{\omega}_B \times I_{cm} \boldsymbol{\omega}_B) \end{bmatrix} \quad (2.27)$$

$$u_i = \frac{1}{\tau_i s + 1} u_{in,1} \quad (2.28)$$

Where  $\Delta \mathbf{l}$  is the offset vector of the center of gravity (CoG) in the body frame of reference. From the model (2.27) the linear and angular accelerations are given, is then

necessary to convert those from the body frame and integrate to obtain the position  $\mathbf{x}_W$  and orientation  $\mathbf{q}_W$  of the quadrotor with the respect to the world frame. Then, by adding the gravity term we have

$$\ddot{\mathbf{x}}_{B,g} = R_{\mathbf{q}_W}(\mathbf{q}_W)^T \cdot \begin{bmatrix} 0 \\ 0 \\ -g \end{bmatrix} + \ddot{\mathbf{x}}_B \quad (2.29)$$

where  $g$  is the gravity constant, about 9.81, and  $R_{\mathbf{q}_W}(\mathbf{q}_W)$  is the rotation matrix built from equation (2.14). To derive the velocity  $\dot{\mathbf{x}}_W$  in the world frame, once again by using the rotation matrix we obtain

$$\dot{\mathbf{x}}_W = R_{\mathbf{q}_W}(\mathbf{q}_W) \cdot \dot{\mathbf{x}}_B \quad (2.30)$$

Instead, for the orientation, we use the results from the paragraph 2.2.1 and we get

$$\dot{\mathbf{q}} = \frac{1}{2} \cdot Q(\boldsymbol{\omega}) \cdot \mathbf{q} \quad (2.31)$$

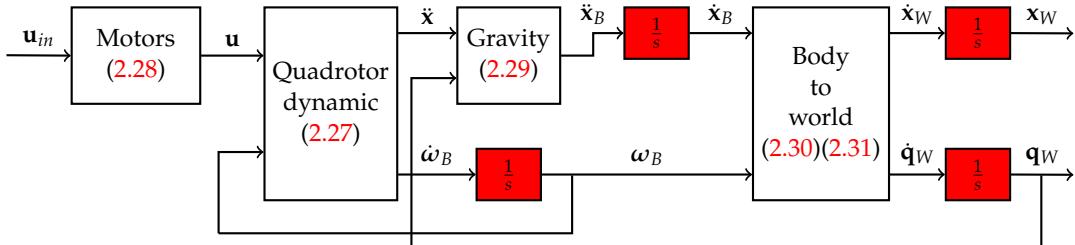


Figure 2.6: Block diagram of the quadrotor's dynamic.

In figure 2.6 is depicted a block diagram of the quadrotor dynamic, from the inputs  $\mathbf{u}_{in}$ , to position  $\mathbf{x}_W$  and orientation  $\mathbf{q}_W$  in the world frame.

### 2.2.3 Adding the rotating platform

Till now, all the model was designed for a standard quadrotor vehicle, what we want to do in this section is to add the model of the rotating platform, necessary for deduce a controller and simulate it.

The movement of the platform, introduces a time variant center of gravity, that is simply modelled with time variant vectors  $\mathbf{l}_i(t)$ , that identify the displacement of the center of the propeller  $i$  with the respect of the CoG. If we know precisely the position of the CoG of the quadrotor (without the moving cart) and the position of the CoG of the cart, the result position can be compute.

In figure 2.7 is illustrated how the resulting CoG change with the position of the cart, is possible to see also the four  $\mathbf{l}_i(t)$  vectors in black dashed line. Then the position of the CoG is

$$\mathbf{p} = \frac{1}{m} \cdot (m_{quad}\mathbf{p}_{quad} + m_{cart}\mathbf{p}_{cart}) \quad (2.32)$$

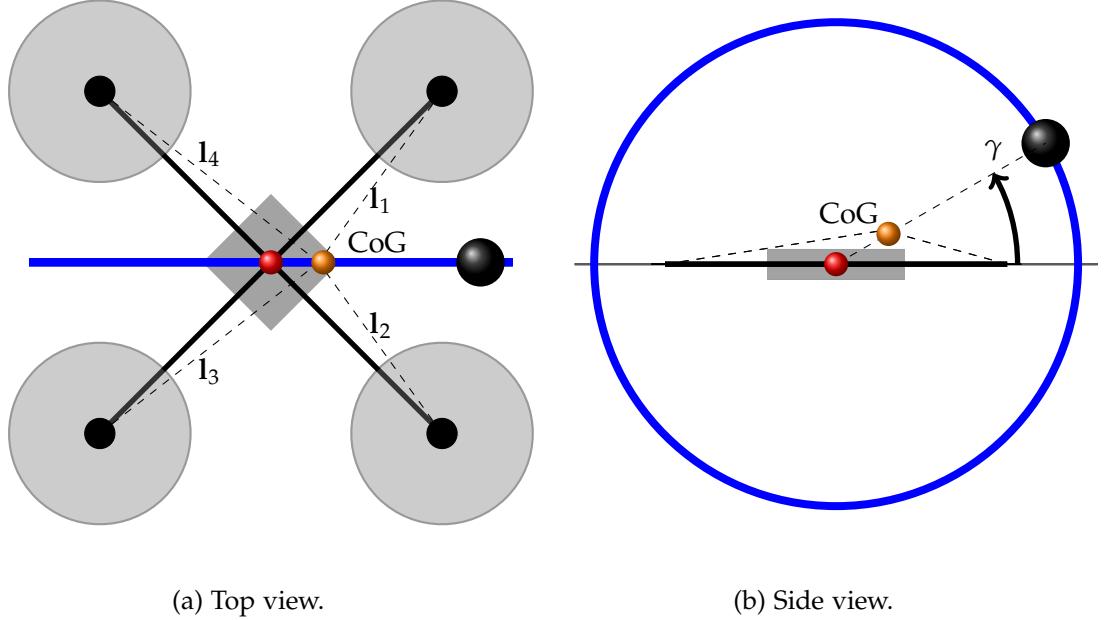


Figure 2.7: Quadrotor with the rotating platform in blue, in red the CoG of the quadrotor and in orange the resulting CoG.

where  $m = m_{quad} + m_{cart}$  is the sum of the mass of the quadrotor plus the mass of the moving cart, then the total mass,  $\mathbf{p}_{quad}$  is the position of the center of gravity of the quadrotor without the cart with the respect to the origin of the body frame (in general the quadrotor frame is not symmetrical) and  $\mathbf{p}_{cart}$  is the position of the CoG of the cart with the respect to the body frame. Then the vectors  $\mathbf{l}_i$  are just the distance between the center of the propeller  $i$  and  $\mathbf{p}$ .

Another difference in using the rotating platform is that the moment of inertia  $I_{cm}$  is not constant, but depend from the position  $\gamma$  of the cart, like in figure 2.7b. This problem can be solved by using the detailed CAD model of the entire vehicle, provided in [3]. From this is possible to deduce the inertia for various position, and then create a simple piecewise model.

The movement of the sensor introduced also a centrifugal force in the vehicle. In particular, if  $\mathbf{p}_{cart}$  is the vector that encode the position of the cart with the respect of the body frame, the Newton's law of motion for the cart in vector form is

$$\mathbf{f} = m_{cart} \mathbf{a} = m_{cart} \frac{d^2 \mathbf{p}_{cart}}{dt^2} \quad (2.33)$$

By twice applying the transformation above from the stationary to the rotating frame, the absolute acceleration of the particle can be written as [10]

$$\begin{aligned} \frac{d^2 \mathbf{p}_{cart}}{dt^2} &= \frac{\partial}{\partial t} \left( \frac{d \mathbf{p}_{cart}}{dt} \right) + \boldsymbol{\omega} \times \left( \frac{d \mathbf{p}_{cart}}{dt} \right) \\ &= \frac{\partial}{\partial t} \left( \frac{\partial \mathbf{p}_{cart}}{\partial t} + \boldsymbol{\omega} \times \mathbf{p}_{cart} \right) + \boldsymbol{\omega} \times \left( \frac{\partial \mathbf{p}_{cart}}{\partial t} + \boldsymbol{\omega} \times \mathbf{p}_{cart} \right) \end{aligned} \quad (2.34)$$

where in this case  $\boldsymbol{\omega}$  is the angular velocity of the cart with the respect to the body frame. Expanding expression (2.34), noting that the chain rule applies to differen-

tiation of cross products, that the cross product is distributive over addition, and coupling with equation (2.33), we have

$$\mathbf{f} = m_{cart} \frac{\partial^2 \mathbf{p}_{cart}}{\partial t^2} + \underbrace{m_{cart} \frac{d\boldsymbol{\omega}}{dt} \times \mathbf{p}_{cart}}_{\text{Euler force}} + \underbrace{2m_{cart} \boldsymbol{\omega} \times \frac{\partial \mathbf{p}_{cart}}{\partial t}}_{\text{Coriolis force}} + \underbrace{m_{cart} \boldsymbol{\omega} \times (\boldsymbol{\omega} \times \mathbf{p}_{cart})}_{\text{centrifugal force}} \quad (2.35)$$

That describe the so called Euler, Coriolis and centrifugal force of the moving platform. To add this to the main model, we just simply need to sum up the vector  $\mathbf{f}$ , divide by  $m_{cart}$  to the equation (2.27), in the first three rows of the matrix, that regard the acceleration of the body frame

$$\begin{bmatrix} \ddot{\mathbf{x}}_B \\ \dot{\boldsymbol{\omega}}_B \end{bmatrix} = \begin{bmatrix} \dots & \frac{A_{F,i}\Omega_{max,i}^2 \mathbf{n}_i}{m} & \dots \\ \dots & I_{cm}^{-1} \left[ (\mathbf{l}_i + \Delta \mathbf{l}) \times A_{F,i}\Omega_{max,i}^2 \mathbf{n}_i - \text{sgn}(\Omega_i) B_{F,i}\Omega_{max,i}^2 \mathbf{n}_i \right] & \dots \\ \dots & I_{cm}^{-1} (\boldsymbol{\omega}_B \times I_{cm} \boldsymbol{\omega}_B) & \mathbf{0} \end{bmatrix} + \begin{bmatrix} \vdots \\ u_i^2 \\ \vdots \end{bmatrix} + \frac{1}{m_{cart}} \begin{bmatrix} \mathbf{f} \\ \mathbf{0} \end{bmatrix} \quad (2.36)$$

## 2.3 Experimental setup

In this section we are going to introduce all the hardware and the software use in this project. Starting from the hardware, we used a main board develop entirely at LTU, the *KFly*<sup>3</sup>.

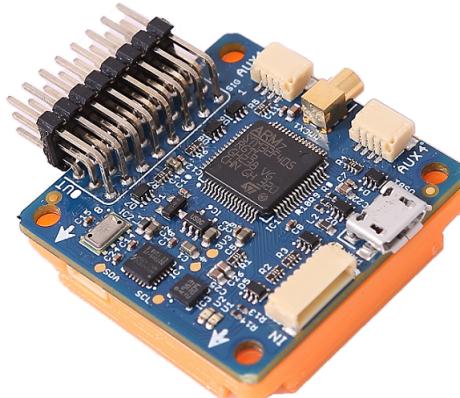


Figure 2.8: KFly board.

In figure 2.8 is reported a picture of the KFly board. It is a small ( $36 \times 36$  mm) but powerfull enough board able to perform all the operations needed during the flight. It is also equipped with differnt sensor, such as an accelerometer and a gyroscope sensors, a magnetometer and a pressure sensors. With the accelerometer is possible to directly measured the acceleration in the body frame, while with the gyroscope is possible to

---

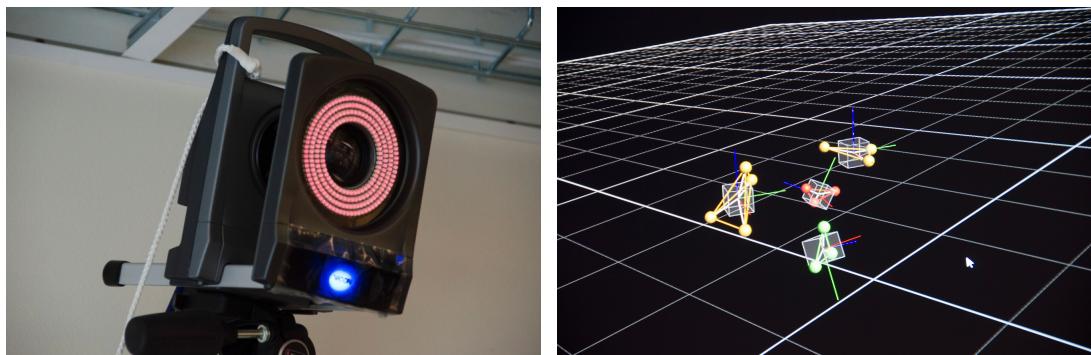
<sup>3</sup><https://gitlab.com/korken89/KFly>

directly measured the angular velocity. These two, combined with the magnetometer form a Inertial Measurement Unit (IMU). It is also equipped with 8 outputs (we will used 4 of them to control the motors) and 4 expansion connectors (3 UARTs and 1 CAN port) for the programming, communication, etcetera. For the communication between the vehicle and the base station, we used the XBee Pro modules<sup>4</sup>.



Figure 2.9: XBee communication modules.

They have a built-in antenna, capable of a transmission range of theoretically 1000 meters. Moreover they have a maximum data rate of 250kbps. Like previously said the UAV is provided with an IMU, but to measure directly the pose of the vehicle other sensors are needed. In particular, to test the control with extremely precision, we used a motion capture system. More precisely we used a Vicon<sup>5</sup> motion capture in the control lab of LTU.



(a) One Vicon camera.

(b) Representation of objects in the Vicon system.

Figure 2.10: Vicon motion capture system.

The system is composed by 20 different cameras, mounted in the perimeter of the lab. By applying a number of markers in the object, is possible to track its position and orientation in the space, with a precision down to the tenth of millimeter.

<sup>4</sup><http://www.digi.com/lp/xbee>

<sup>5</sup><https://www.vicon.com/>

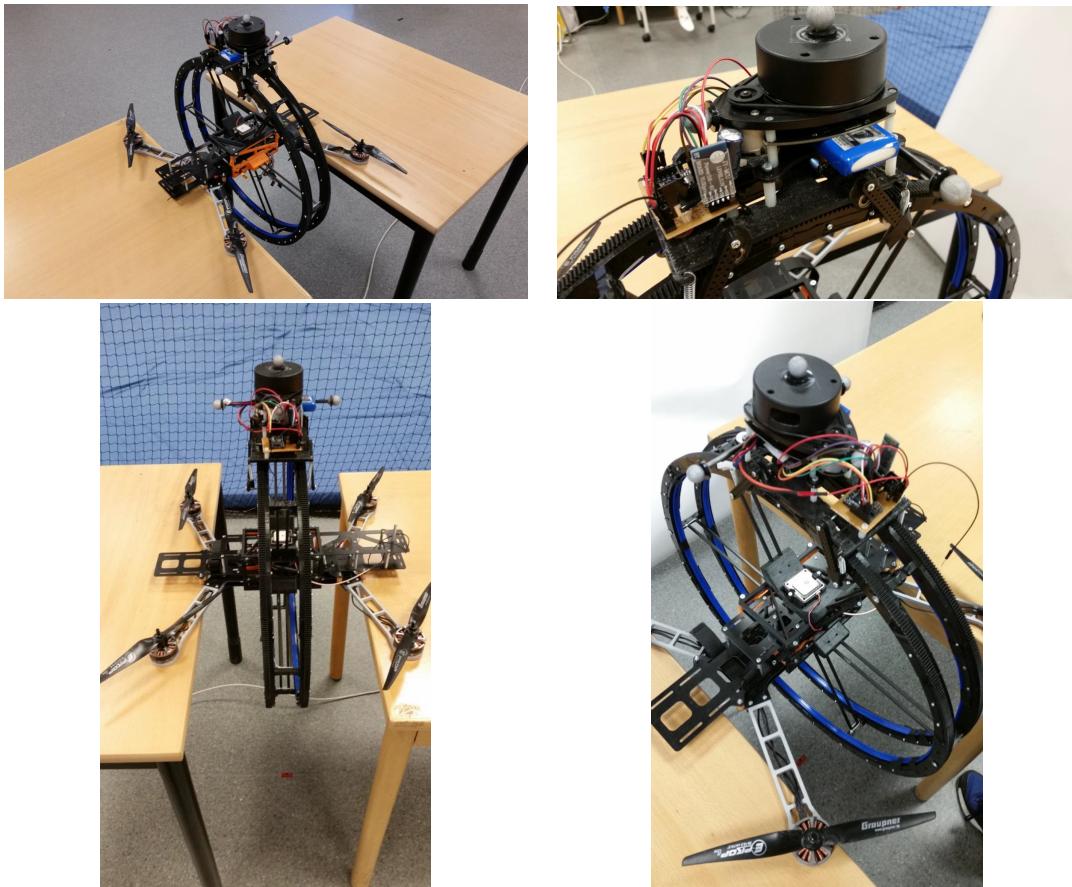


Figure 2.11: First prototype of the Prometheus mapping drone.

In figure 2.11 there are some pictures of the first prototype of the Prometheus mapping drone. In particular, it is possible to see all the electronics and the Vicon markers on top and on the side of the Lidar sensor.

Moving to the software part, all the simulations are made in a *MATLAB* and *SIMULINK* environment. Instead, for the real application, we choose to use *ROS*, the Robotic Operating System<sup>6</sup> and then to rewrite all the code in C++. *ROS* is an open source project that is a collection of software frameworks for robot software development, providing operating system-like functionality on a heterogeneous computer cluster. *ROS* provides standard operating system services such as hardware abstraction, low-level device control, implementation of commonly used functionality, message-passing between processes, and package management. It is very popular nowadays in a robotics project and there is a big worldwide community. However, the main problem was that the KFly can't run *ROS* onboard. We solve this problem by using a laptop with *ROS* installed to run the control system, then send the control inputs to the KFly via XBee. By using this strategy we encounter problems with the bandwidth of the XBee, reaching its saturation limit, if we run the control algorithm at more than 50 Hertz (100 Hz is quite standard in quadrotor application). A possible solution could be to use another low cost board, such as the *Odroid c2*<sup>7</sup>, that can use high speed wi-fi link for the communication, and then practically unlimited bandwidth, and can run *ROS*.

<sup>6</sup><http://www.ros.org/>

<sup>7</sup><http://www.hardkernel.com/main/main.php>

# 3

## System identification

In this chapter, is about to be addressed an important part of this project. Since the model of the previous chapter is depending from many parameters, is necessary to identificate them, to be able to design an appropriate controller. A Kalman Filter approach will be used, based from the work [9].

### 3.1 System simplification and linear approximation

Starting from the model deducted in section 2.2.2

$$\begin{bmatrix} \ddot{\mathbf{x}}_B \\ \dot{\boldsymbol{\omega}}_B \end{bmatrix} = \begin{bmatrix} \dots & \frac{A_{F,i}\Omega_{max,i}^2 \mathbf{n}_i}{m} \\ \dots & I_{cm}^{-1} \left[ (\mathbf{l}_i + \Delta \mathbf{l}) \times A_{F,i}\Omega_{max,i}^2 \mathbf{n}_i - \text{sgn}(\Omega_i) B_{F,i}\Omega_{max,i}^2 \mathbf{n}_i \right] \end{bmatrix} \begin{bmatrix} \vdots \\ u_i^2 \\ \vdots \end{bmatrix} + \\ + \begin{bmatrix} \mathbf{0} \\ I_{cm}^{-1} (\boldsymbol{\omega}_B \times I_{cm} \boldsymbol{\omega}_B) \end{bmatrix} \quad (3.1)$$

$$u_i = \frac{1}{\tau_i s + 1} u_{in,1} \quad (3.2)$$

we need to do some simplification. In particular, by assuming that all engines have the same parameters, is possible to rewrite these parameters as follows

$$A_{F,i}\Omega_{max,i}^2 \approx A_F \quad (3.3)$$

$$B_{F,i}\Omega_{max,i}^2 \approx B_F \quad (3.4)$$

$$\tau_i \approx \tau \quad (3.5)$$

Moreover the term  $I_{cm}^{-1}(\boldsymbol{\omega} \times I_{cm}\boldsymbol{\omega}_B)$  can be neglected [9]. This can be easily seen simply by simulating the mathematical model with and without the term, the differences are very small.

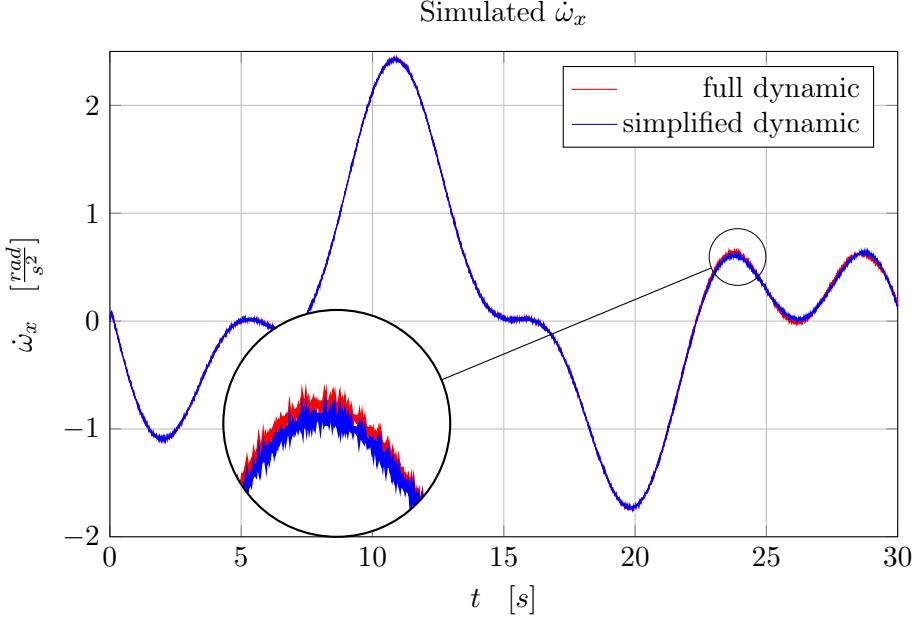


Figure 3.1: Simulation of the dynamic with and without the term  $I_{cm}^{-1}(\boldsymbol{\omega} \times I_{cm}\boldsymbol{\omega}_B)$ .

Another simplification, is that the inertia matrix  $I_{cm}$  is a diagonal matrix,  $I_{cm} = \text{diag}(I_{xx}, I_{yy}, I_{zz})$ . This is generally true in a standard quadrotor but is not so immediate for the vehicle of this project. However, if we align the  $x$  axis with the orientation of the circular structure, we obtain a inertia matrix almost diagonal; what makes the matrix "less diagonal" is the position of the cart. However, the mass of the sensor is not sufficiently big to modify enough the matrix and this assumption is valid also here. Of course, in different applications, where the mass of the quadrotor and the mass of the sensor are more similar, a different approach is required. Another non linearity is in the inputs, since the model require the square of these. A solution of this problem proposed in [9] is to rewrite equation (3.2) with the square of the control inputs. This effectively moves the squared control signal from the force and torque equations to the input. This representation keeps the static relationship but will affect the dynamics of the first order system, but is assumed that a first order system still captures the majority of the dynamics. In conclusion, the approximate linear model is

$$\begin{bmatrix} \ddot{\mathbf{x}}_B \\ \dot{\boldsymbol{\omega}}_B \end{bmatrix} = \begin{bmatrix} \cdots & \frac{A_F \mathbf{e}_3}{m} \\ \cdots & I_{cm}^{-1} \left[ (\mathbf{l}_i + \Delta \mathbf{l}) \times A_F \mathbf{e}_3 - \text{sgn}(\Omega_i) B_F \mathbf{e}_3 \right] \end{bmatrix} \begin{bmatrix} \vdots \\ u_i \\ \vdots \end{bmatrix} \quad (3.6)$$

$$u_i = \frac{1}{\tau s + 1} u_{in,i}^2 \quad (3.7)$$

where instead of  $\mathbf{n}_i$  there is  $\mathbf{e}_3$  because in the structure of this particular vehicle, the propellers are mounted parallel to the ground and then with a force vector aligned to  $\mathbf{e}_3 = [0 \ 0 \ 1]^T$ .

### 3.2 Quadrotor parameters

From the simplified model of equations (3.6) and (3.7), the identifiable parameters are

$$\boldsymbol{\beta} = \left[ \frac{A_F}{m} \quad \frac{A_F}{I_{xx}} \quad \frac{A_F}{I_{yy}} \quad \frac{A_F}{I_{zz}} \quad \frac{B_F}{I_{zz}} \quad \Delta l_x \quad \Delta l_y \right]^T, \quad \tau \quad (3.8)$$

Then, is possible to rewrite the linear model in a more compact form:

$$\begin{bmatrix} \dot{\mathbf{x}}_B \\ \dot{\boldsymbol{\omega}}_B \end{bmatrix} = \begin{bmatrix} L(\boldsymbol{\beta}_1) \\ A(\boldsymbol{\beta}) \end{bmatrix} \mathbf{u} \quad (3.9)$$

Under the assumption of the sampling rate to be much faster than the dynamics <sup>1</sup>, equation (3.7) is implemented as discrete-time first order system, and the parameters are modeled as integrated white noise, which gives the following prediction equations

$$\boldsymbol{\omega}_k = \boldsymbol{\omega}_{k-1} + \Delta t A(\boldsymbol{\beta}_{k-1}) \mathbf{u}_{k-1} \quad (3.10)$$

$$\mathbf{u}_k = \frac{\tau_{k-1}}{\Delta t + \tau_{k-1}} \mathbf{u}_{k-1} + \frac{\Delta t}{\Delta t + \tau_{k-1}} \mathbf{u}_{in,k}^2 \quad (3.11)$$

$$\boldsymbol{\beta}_k = \boldsymbol{\beta}_{k-1} \quad (3.12)$$

$$\tau_k = \tau_{k-1} \quad (3.13)$$

where  $\mathbf{u}_k$  and  $\mathbf{u}_{in,k}$  are the inputs at time instant  $k$  expressed in a vectorial way,  $\Delta t$  is the sampling period and  $\bullet^2$  is the element-wise square of a vector.

### 3.3 Kalman filter

A Kalman filter approach is chose for this project since it has good result in this kind of applications. Of course, better performances can be obtained with specific strategies for non linear systems [11], but these methods are in general much more complicated and require much more computational effort, especially if is necessary to estimate the parameters online.

Now, is possible to use the standard Kalman filter equations [12] to develop an online identification algoritm as follow.

The augmented state  $\mathbf{x}_{est}$  is

$$\mathbf{x}_{est} = [\boldsymbol{\omega}_B \quad \mathbf{u}_{in} \quad \boldsymbol{\beta} \quad \tau]^T \in \mathbb{R}^{15} \quad (3.14)$$

The initial values of  $\boldsymbol{\omega}_B$  and  $\mathbf{u}_{in}$  are know, so the state is initialized with these. Moreover, due to the parameters  $\boldsymbol{\beta}$  and  $\tau$  having a constraint to being positive, they are implemented as  $\exp(\boldsymbol{\beta})$  and  $\exp(\tau)$  to force positive results from the estimation, while the  $\Delta l$  are constrained to be within the propellers (the length of the arms is set to be

---

<sup>1</sup>in this case, thanks to the performance of the onboard electronics, the sampling rate is equal to 222 Herz.

equal to one, since the correct length is not necessary for the identification) which is implemented using a zero centered logistic function

$$\frac{2}{1 - \exp(-\Delta l)} - 1 \quad (3.15)$$

With the augmented state it is possible to write a new state space system in discrete time with matrix  $A_{est}$ <sup>2</sup>

$$\begin{aligned}
 A_{\omega, \mathbf{u}_{in}} &= \begin{bmatrix} 2\Delta t e^{\beta_2} (\Delta l_y - 1) \cdot \mathbf{u}^T \\ -2\Delta t e^{\beta_3} (\Delta l_x + 1) \cdot \mathbf{u}^T \\ -\text{sgn}(\Omega_i) 2\Delta t e^{\beta_4} \cdot \mathbf{u}^T \end{bmatrix} \in \mathbb{R}^{3 \times 4} \\
 A_{\omega, \beta_1} &= \mathbf{0}_{3 \times 1} \in \mathbb{R}^{3 \times 1} \\
 A_{\omega, \beta_2} &= \left[ \Delta t e^{\beta_2} \left( \Delta l_y \sum_{i=1}^4 u_i^2 - u_1^2 - u_2^2 + u_3^2 + u_4^2 \right) \quad 0 \quad 0 \right]^T \in \mathbb{R}^{3 \times 1} \\
 A_{\omega, \beta_3} &= \left[ 0 \quad -\Delta t e^{\beta_3} \left( \Delta l_y \sum_{i=1}^4 u_i^2 + u_1^2 - u_2^2 - u_3^2 + u_4^2 \right) \quad 0 \right]^T \in \mathbb{R}^{3 \times 1} \\
 A_{\omega, \beta_4} &= \left[ 0 \quad 0 \quad -\Delta t e^{\beta_4} \sum_{i=1}^4 \text{sgn}(\Omega_i) u_i^2 \right]^T \in \mathbb{R}^{3 \times 1} \\
 A_{\omega, \beta_5} &= [0 \quad 0 \quad \Delta t]^T \in \mathbb{R}^{3 \times 1} \\
 A_{\omega, \beta_{6,7}} &= \begin{bmatrix} 0 & \Delta t e^{\beta_2} \sum_{i=1}^4 u_i^2 \\ -\Delta t e^{\beta_3} \sum_{i=1}^4 u_i^2 & 0 \\ 0 & 0 \end{bmatrix} \in \mathbb{R}^{3 \times 2} \\
 A_{\omega, \beta} &= [A_{\omega, \beta_1} \mid A_{\omega, \beta_2} \mid A_{\omega, \beta_3} \mid A_{\omega, \beta_4} \mid A_{\omega, \beta_5} \mid A_{\omega, \beta_{6,7}}] \in \mathbb{R}^{3 \times 7} \\
 A_{\mathbf{u}_{in}} &= \left( 1 - \frac{\Delta t}{\Delta t + e^\tau} \right) \cdot I_4 \in \mathbb{R}^{4 \times 4} \\
 A_{est, k} &= \begin{bmatrix} I_3 \mid A_{\omega, \mathbf{u}_{in}} \mid A_{\omega, \beta} \mid \mathbf{0}_{3 \times 1} \\ \hline \mathbf{0}_{4 \times 3} \mid A_{\mathbf{u}_{in}} \mid \mathbf{0}_{4 \times 8} \\ \hline \mathbf{0}_{8 \times 7} \mid I_8 \end{bmatrix} \in \mathbb{R}^{15 \times 15}
 \end{aligned}$$

and then use the Kalman filter equations in a recursive way [12]

---

<sup>2</sup>For notation,  $\mathbf{0}_{a \times b}$  is equal to a zero matrix with  $a$  rows and  $b$  columns,  $\mathbf{1}_{a \times b}$  is equal to a ones matrix with  $a$  rows and  $b$  columns,  $I_a$  is the identity matrix of dimension  $a \times a$ , and the vector  $\mathbf{x}_{est, a:b}$  are the entries from  $a$  to  $b$  of the augmented state (is implicit that is consider at time  $k$ )

$$\begin{aligned}
P_k &= A_{est,k} \cdot P_{k-1} \cdot A_{est,k}^T + Q && \in \mathbb{R}^{15 \times 15} \\
H_k &= \left[ \begin{array}{c|c|c|c|c} I_3 & & \mathbf{0}_{3 \times 12} & & \\ \hline \mathbf{0}_{1 \times 3} & 2e^{\beta_1} \cdot \mathbf{u}^T & \mathbf{0}_{1 \times 2} & e^{\beta_1} \sum_{i=1}^4 u_i^2 & \mathbf{0}_{1 \times 5} \end{array} \right] && \in \mathbb{R}^{4 \times 15} \\
S_k &= H_k \cdot P_k \cdot H_k^T + R && \in \mathbb{R}^{4 \times 4} \\
K_k &= P_k \cdot H_k^T \cdot S_k^{-1} && \in \mathbb{R}^{15 \times 4} \\
P_k &= (I_{15} - K_k \cdot H_k) \cdot P_{k-1} && \in \mathbb{R}^{15 \times 15} \\
\mathbf{x}_{est,k} &= \mathbf{x}_{est,k-1} + K_k \cdot \left( \begin{bmatrix} \boldsymbol{\omega} \\ \ddot{x}_z \end{bmatrix} - \begin{bmatrix} \mathbf{x}_{est,1:3} \\ e^{\beta_1} \cdot \mathbf{1}_{1 \times 4} \cdot \mathbf{x}_{est,4:7}^2 \end{bmatrix} \right) && \in \mathbb{R}^{15 \times 1}
\end{aligned}$$

where  $P_k$  is the state update covariance matrix based on model,  $H_k$  maps the measurement to the states,  $S_k$  is the update measurement covariance,  $K_k$  is the update Kalman gain,  $Q \in \mathbb{R}^{15 \times 15}$  is the fixed covariance matrix and  $R \in \mathbb{R}^{4 \times 4}$  the fixed measurement covariance matrix. Both  $Q$  and  $R$  are diagonal matrices.

### 3.4 Results

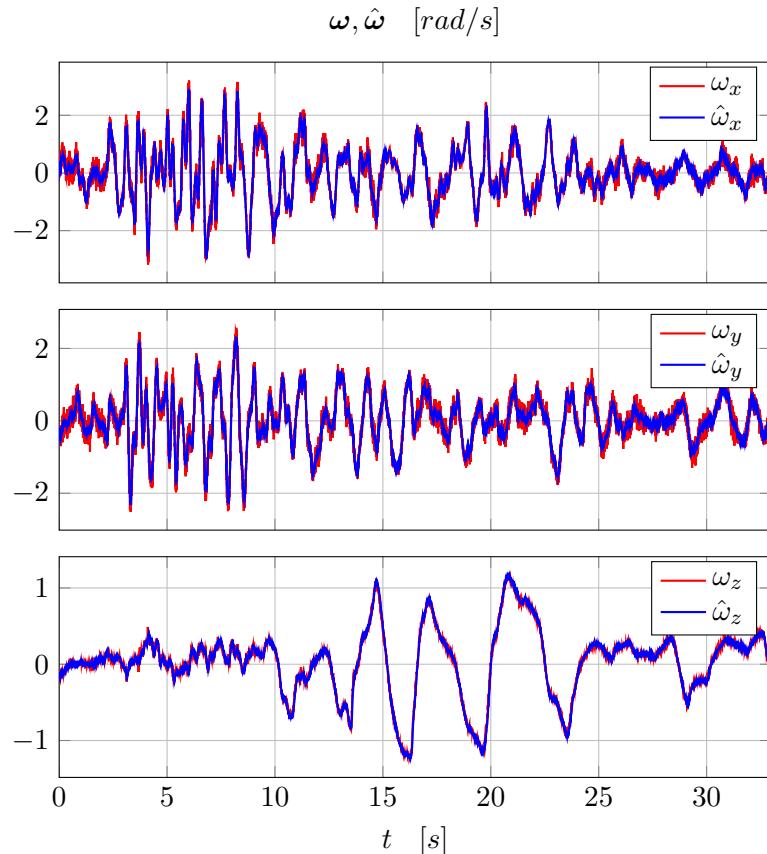


Figure 3.2: Measured and estimated angular rate,  $\boldsymbol{\omega}_B$  and  $\hat{\boldsymbol{\omega}}_B$ .

The estimator needs to be setup with specific process and measurement covariance  $Q$

and  $R$ , and the starting state covariance  $P_0$ . The values for  $Q$  and  $P_0$  were found simply with a trial and error procedure, while  $R$  was taken from the noise densities of the measured signals. In particular we measured a steady state position of the quadrotor, record the acceleration in the z axis  $\ddot{x}_{B,z}$  and the angular rate  $\omega_B$ , then by analyzing these data, a noise variance was extracted. Moreover, since in the augmented state  $\mathbf{x}_{est}$  is present also an estimation of the angular rate  $\hat{\omega}$ , to evaluate the quality of the estimation was also compare it with the measured angular rate. In this case the initial values of the state  $\mathbf{x}_{est}$  were chose to be considerably different from a real value, just to show the performance of the estimator.

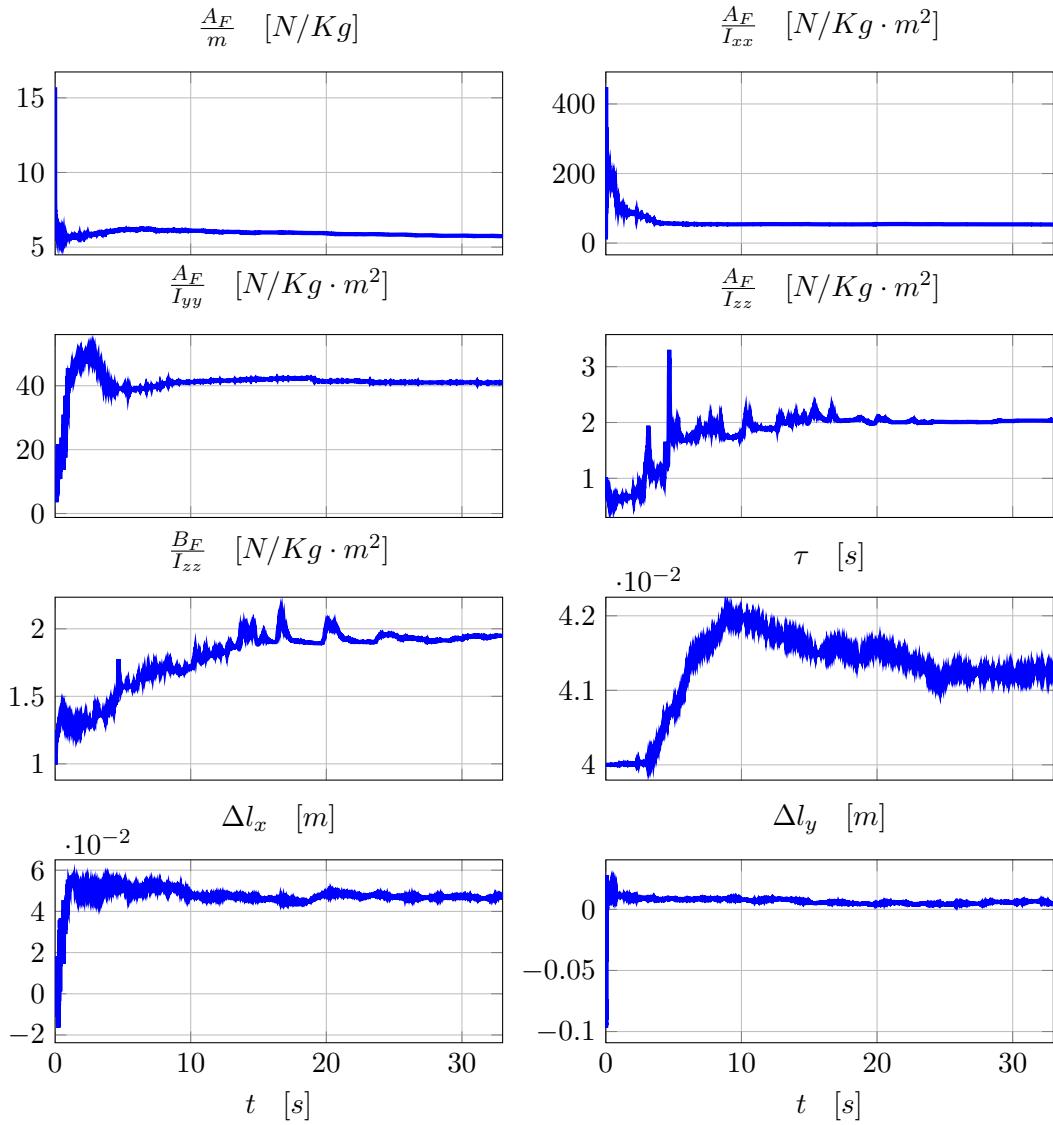


Figure 3.3: Estimated parameters  $\beta$  and  $\tau$ .

As is possible to see in figure 3.2, the estimation of the angular rate works pretty well from the begining for all the three axis. In figure 3.3 are instead plotted the estimations of all parameters  $\beta$  and  $\tau$ . The identification was performed with the cart fixed in one position. Is possible to see that for almost all parameters there is convergence after about 15 seconds, while for the parameter  $\frac{B_F}{I_{zz}}$  is necessary more time. This can be

explain by observing the angular rate, in particular note that  $\omega_z$  is almost zero for about the first 10 seconds, due to the particular trajectory of the vehicle. Of course, is not possible to perform system identification without excitation of the system and that's why the parameters depending on  $\omega_z$  require more time to converge.  
In conclusion the algorithm has been shown to work quite well, and for this application is not then necessary to use more sophisticated techniques.



# 4

## Trajectories generator

An important aspect of this project is the path planning, because the aim is to map and navigate in an environment without a priori information and in complete autonomous. The study of path planning algorithms and the sensors' fusion to obtain the pose of the vehicle is not part of this thesis, however part of this thesis is to generate a trajectory for the UAV based on the output of a path planning algorithm. In particular, usually exploration algorithms provide only setpoints, not full trajectories [13] [14]. It is then necessary to provide a tool to generate possible trajectories with constraints in the environment and in the dynamic of the vehicle. In this project we implemented the solution proposed in the works [15] and [16], where the authors, after providing model and control, generate a trajectory composed by piecewise polynomial functions.

### 4.1 Trajectory definition

In this work, a *setpoint*  $\sigma_d$  is defined as a position in the space,  $\mathbf{x}_d$ , and a yaw angle,  $\psi_d$ , since in the next section we will control four degrees of freedom of the UAV, the position in the space and the yaw angle. We consider the problem of navigate through  $m$  setpoints at specific time. A trivial trajectory is one that interpolate the setpoints using straight lines. However, such trajectory is inefficient because it require the quadrotor to come to a stop at each setpoint. This method generates trajectories that smoothly transition through the setpoints at given times. We write down a trajectory as piecewise polynomial functions of order  $n$  over  $m$  time intervals

$$\sigma_d(t) = \begin{cases} \sum_{i=0}^n \sigma_{d,i,1} t^i & t_0 \leq t < t_1 \\ \sum_{i=0}^n \sigma_{d,i,2} t^i & t_1 \leq t < t_2 \\ \vdots \\ \sum_{i=0}^n \sigma_{d,i,m} t^i & t_{m-1} \leq t < t_m \end{cases} \quad (4.1)$$

where  $\sigma_{d,i,j}$  is the coefficient of orden  $i$  of the trajectory piece  $j$  and  $t_k$  is the time that the vehicle has to reach setpoint  $k$ ;  $i \in [0 \ n]$ ,  $j \in [1 \ m]$ ,  $k \in [1 \ m]$ . The interest is then to minimize a cost function which can be written using these piecewise polynomial.

$$\begin{aligned} \min \quad & \int_{t_0}^{t_m} \mu_x \left\| \frac{d^{k_x} \mathbf{x}_d}{dt^{k_x}} \right\|^2 + \mu_\psi \left( \frac{d^{k_\psi} \psi_d}{dt^{k_\psi}} \right)^2 dt \\ \text{subject to} \quad & \sigma_d(t_i) = \sigma_{d,i}, \quad i = 0, \dots, m \\ & \frac{d^p x_d}{dt^p} \Big|_{t=t_j} = 0, \quad j = 0, m; \quad p = 1, \dots, k_r \\ & \frac{d^p y_d}{dt^p} \Big|_{t=t_j} = 0, \quad j = 0, m; \quad p = 1, \dots, k_r \\ & \frac{d^p z_d}{dt^p} \Big|_{t=t_j} = 0, \quad j = 0, m; \quad p = 1, \dots, k_r \\ & \frac{d^p \psi_d}{dt^p} \Big|_{t=t_j} = 0, \quad j = 0, m; \quad p = 1, \dots, k_\psi \end{aligned} \quad (4.2)$$

where  $\mu_x$  and  $\mu_\psi$  are constants that make the integrand nondimensional. Here  $\sigma_d = [x_d \ y_d \ z_d \ \psi_d]^T$  and  $\sigma_{d,i} = [x_{d,i} \ y_{d,i} \ z_{d,i} \ \psi_{d,i}]^T$ . We also assume that  $t_0 = 0$  without loss of generality. The first constraint indicates that the result trajectory has to pass through the desire setpoints, while the rest of the constraints impose that all the derivatives at the initial and final point have to be zero (is also possible to set them to a specific value if necessary). By using the same choice of [17], we decide to minimize the snap for the position ( $k_x = 4$ ) and the second derivative of the yaw angle ( $k_\psi = 2$ ). Now we want to formulate the trajectory generation problem as an optimization of a functional but in a finite dimensional setting. This will keep the computational effort very small to guarantee a real time application. In order to do this, we first write the constants  $\sigma_{d,i,j} = [x_{d,i,j} \ y_{d,i,j} \ z_{d,i,j} \ \psi_{d,i,j}]^T$  as a  $4 \cdot n \cdot m \times 1$  vector  $\mathbf{c}$  with decision variables  $\{x_{d,i,j}, y_{d,i,j}, z_{d,i,j}, \psi_{d,i,j}, i \in [0 \ n], j \in [0 \ m]\}$ . The trajectory generation preoblem (4.3) can be written in the form of a quadratic program (QP):

$$\begin{aligned} \min \quad & \mathbf{c}^T H \mathbf{c} + f^T \mathbf{c} \\ \text{subject to} \quad & A \mathbf{c} \leq \mathbf{b} \\ & A_{eq} \mathbf{c} = \mathbf{b}_{eq} \end{aligned} \quad (4.3)$$

where the objective function will incorporate the minimization of the functional while the constraint can be used to satisfy constraints in the trajectory and its derivatives. A specification of an initial condition, final condition, or intermediate condition on any derivative of the trajectory (e.g.  $\frac{d^k x_d}{dt^k}$ ) can be written as a row of the constraints  $A_{eq} \mathbf{c} = \mathbf{b}_{eq}$ . If conditions do not need to be specified exactly then they can be represented with the inequality constraint  $A \mathbf{c} \leq \mathbf{b}$ .

Moreover, to simplify the problem, can be notice that in the cost function of equation (4.3), the four dimensions are independent, this means that the problem can be split in four different problems for each dimension. In such a way, the construction of the

quadratic problem vectors and matrices will be considerably more simple. Furthermore, it is possible to assume that each setpoint starts from  $t_0 = 0$  and ends at  $t_j = 1$ . This is because if we define a new time variable such as

$$\tau = \frac{t - t_{j-1}}{t_j - t_{j-1}} \quad (4.4)$$

the new one dimension position at time  $\tau$  becomes

$$x(\tau) = x\left(\frac{t - t_{j-1}}{t_j - t_{j-1}}\right) \quad (4.5)$$

and its derivatives

$$\begin{aligned} \frac{d}{dt}x(t) &= \frac{d}{dt}x(\tau) \\ &= \frac{d}{d\tau} \frac{d\tau}{dt} x(\tau) \\ &= \frac{1}{t_j - t_{j-1}} \frac{d}{d\tau} x(\tau) \\ &\dots \\ \frac{d^k}{dt^k}x(t) &= \frac{1}{(t_j - t_{j-1})^k} \frac{d^k}{d\tau^k} x(\tau) \end{aligned}$$

We can thus solve for each piece of any piece-wise trajectory from  $\tau = 0$  to 1, then scale to any  $t_0$  to  $t_j$ .

## 4.2 Optimization of a trajectory between two setpoints

To make the derivation simpler, it is better to start the optimization problem with only two setpoints and in only one dimension. In particular, the cost function becomes

$$J = \int_{t_0}^{t_1} \left| \left| \frac{d^k x(t)}{dt^k} \right| \right|^2 dt = \mathbf{c}^T H_{(t_0, t_1)} \mathbf{c} \quad (4.6)$$

subject to  $A\mathbf{c} = \mathbf{b}$

We can instead look for the non-dimensionalized trajectory  $x(\tau) = c_n \tau^n + c_{n-1} \tau^{n-1} + \dots + c_1 \tau + c_0$  where  $\tau = \frac{t - t_0}{t_1 - t_0}$ . Note that this makes  $\tau$  range from 0 to 1. Let  $\mathbf{c} = [c_n \ c_{n-1} \ \dots \ c_1 \ c_0]^T$ . We can write the cost function  $J$  in terms of the non-dimensionalized trajectory  $x(\tau)$ :

$$\begin{aligned}
J &= \int_{t_0}^{t_1} \left\| \frac{d^k x(t)}{dt} \right\|^2 dt \\
&= \int_0^1 \left\| \frac{1}{(t_1 - t_0)^k} \frac{d^k x(\tau)}{d\tau} \right\|^2 d(\tau(t_1 - t_0) + t_0) \\
&= \frac{t_1 - t_0}{(t_1 - t_0)^{2k}} \int_0^1 \left\| \frac{d^k x(\tau)}{d\tau} \right\|^2 d\tau \\
&= \frac{1}{(t_1 - t_0)^{2k-1}} \mathbf{c}^T H_{(0,1)} \mathbf{c} \\
&= \mathbf{c}^T \left( \frac{1}{(t_1 - t_0)^{2k-1}} H_{(0,1)} \right) \mathbf{c}
\end{aligned} \tag{4.7}$$

Thus, we want to minimize the cost function

$$J = \mathbf{c}^T \left( \frac{1}{(t_1 - t_0)^{2k-1}} H_{(0,1)} \right) \mathbf{c} \tag{4.8}$$

subject to  $A\mathbf{c} = \mathbf{b}$

To find  $H'_{(0,1)}$ , when  $\mathbf{c}' = [c_0 \ c_1 \ \dots \ c_{n-1} \ c_n]^T$ , we can find  $H'_{(0,1)}$  with:

$$H'[i,j]_{(t_0,t_1)} = \begin{cases} \prod_{z=0}^{k-1} (i-z)(j-z)^{\frac{t_1^{i+j-2k+1} - t_0^{i+j-2k+1}}{i+j-2k+1}} & i \geq k \wedge j \geq k \\ 0 & i < k \vee j < k \end{cases} \tag{4.9}$$

$i = 0, \dots, n, \quad j = 0, \dots, n$

However,  $\mathbf{c} = [c_n \ c_{n-1} \ \dots \ c_1 \ c_0]^T$ . Reflecting  $H'$  from equation (4.9) horizontally and vertically will give the desire  $H$  for the form of  $\mathbf{c}$  we desire. The function to minimize is then  $\left( \frac{1}{(t_1 - t_0)^{2k}} H_{(0,1)} \right)$ .

To find  $A$

$$A\mathbf{c} = \mathbf{b}$$

$$\begin{bmatrix} A(t_0) \\ A(t_1) \end{bmatrix} \mathbf{c} = \begin{bmatrix} x(t_0) \\ \vdots \\ x^{(k-1)}(t_0) \\ x(t_1) \\ \vdots \\ x^{(k-1)}(t_1) \end{bmatrix}$$

Note that  $A\mathbf{c}$  only contains rows where constraints are specified, if a condition is unconstrained just omit a row. Assuming that every condition is constrained, the general form of  $A$  is:

$$A[i, j](t) = \begin{cases} \prod_{z=0}^{i-1} (n-z-j)t^{n-j-i} & n-j \geq i \\ 0 & n-j < i \end{cases} \quad (4.10)$$

$i = 0, \dots, r-1, \quad j = 0, \dots, n$

where  $A[i, j]$  represents the  $(n-j)$ th coefficient of the  $i$ th derivative. In the non-dimensionalized case, we have  $\tau_0 = 0$  and  $\tau_1 = 1$ :

$$\begin{bmatrix} A(\tau_0) \\ A(\tau_1) \end{bmatrix} \mathbf{c} = \begin{bmatrix} x(t_0) \\ \vdots \\ (t_1 - t_0)^{k-1} x^{(k-1)}(t_0) \\ x(t_1) \\ \vdots \\ (t_1 - t_0)^{k-1} x^{(k-1)}(t_1) \end{bmatrix} \quad (4.11)$$

### 4.3 Optimization of a trajectory between $m+1$ setpoints

In this section, by recalling what we study in the previous section, is possible to derive the equations to optimize a trajectory for an arbitrary number of setpoints. In particular, we seek the piece-wise trajectory:

$$x(t) = \begin{cases} x_1(t), & t_0 \leq t < t_1 \\ x_2(t), & t_1 \leq t < t_2 \\ \vdots \\ x_m(t), & t_{m-1} \leq t < t_m \end{cases} \quad (4.12)$$

and continue to minimize the cost function

$$J = \int_{t_0}^{t_m} \left\| \frac{d^k x(t)}{dt^k} \right\|^2 dt = \mathbf{c}^T H_{(t_0, t_m)} \mathbf{c} \quad (4.13)$$

subject to  $A\mathbf{c} = \mathbf{b}$

and again look for the non-dimensionalized trajectory

$$x(\tau) = \begin{cases} x_1(\tau) = c_{1,n}\tau^n + \dots + c_{1,0}, & t_0 \leq t < t_1, \quad \tau = \frac{t-t_0}{t_1-t_0} \\ x_m(\tau) = c_{m,n}\tau^n + \dots + c_{m,0}, & t_{m-1} \leq t < t_m, \quad \tau = \frac{t-t_{m-1}}{t_m-t_{m-1}} \\ 0 \leq \tau < 1 \end{cases} \quad (4.14)$$

Let  $\mathbf{c}_z = [c_{z,n} \ c_{z,n-1} \ \dots \ c_{z,1} \ c_{z,0}]^T$  and  $\mathbf{c} = [\mathbf{c}_1^T \ \mathbf{c}_2^T \ \dots \ \mathbf{c}_m^T]^T$ . Each piece of the trajectory is optimized individually between  $\tau_0 = 0$  and  $\tau_1 = 1$ . We want to minimize:

$$\begin{aligned}
J &= \int_{t_0}^{t_m} \left\| \frac{d^k x(t)}{dt} \right\|^2 dt \\
&= \sum_{z=1}^m \int_{t_{z-1}}^{t_z} \left\| \frac{d^k x_z(t)}{dt} \right\|^2 dt \\
&= \sum_{z=1}^m \int_0^1 \frac{t_z - t_{z-1}}{(t_z - t_{z-1})^{2k}} \left\| \frac{d^k x_z(\tau)}{d\tau} \right\|^2 d\tau \\
&= \sum_{z=1}^m \mathbf{c}_z^T \frac{1}{(t_z - t_{z-1})^{2k-1}} H_{(0,1)} \mathbf{c}_z \\
&= \mathbf{c}^T H \mathbf{c}
\end{aligned} \tag{4.15}$$

subject to  $A\mathbf{c} = \mathbf{b}$

To find  $H$ , we recall that for each  $\mathbf{c}'_z = [c_{z,0} \ c_{z,1} \ \dots \ c_{z,n-1} \ c_{z,n}]^T$ , where  $z = 1, \dots, m$ ,  $H'_{(0,1)}$  is given by equation (4.9). Since  $\mathbf{c}_z = [c_{z,n} \ c_{z,n-1} \ \dots \ c_{z,1} \ c_{z,0}]^T$ , refletting  $H'$  horizontally and vertically will give the desire  $H$  for the form of  $\mathbf{c}_k$ . Is then possible to create the block diagonal matrix  $H$

$$H = \begin{bmatrix} \frac{1}{(t_1 - t_0)^{2k-1}} H_{(0,1)} & \cdots & 0 & 0 \\ \cdots & \cdots & \cdots & \vdots \\ \cdots & 0 & \frac{1}{(t_{m-1} - t_{m-2})^{2k-1}} H_{(0,1)} & 0 \\ 0 & \cdots & 0 & \frac{1}{(t_m - t_{m-1})^{2k-1}} H_{(0,1)} \end{bmatrix} \tag{4.16}$$

To find  $A$ , first, we need to account for endpoint constraints, in the non-dimensionalized case:

$$\begin{aligned}
A_{\text{endpoint}} \mathbf{c} &= \mathbf{b}_{\text{endpoint}} & (4.17) \\
\begin{bmatrix} A(\tau_0) & 0 & \dots & 0 \\ A(\tau_1) & 0 & \dots & 0 \\ 0 & A(\tau_0) & \dots & 0 \\ 0 & A(\tau_1) & \dots & 0 \\ \vdots & 0 & \dots & \vdots \\ 0 & \dots & 0 & A(\tau_0) \\ 0 & \dots & 0 & A(\tau_1) \end{bmatrix} \mathbf{c} &= \begin{bmatrix} x_1(t_0) \\ \vdots \\ (t_1 - t_0)^{k-1} x_1^{(k-1)}(t_0) \\ x_1(t_1) \\ \vdots \\ (t_1 - t_0)^{k-1} x_1^{(k-1)}(t_1) \\ \vdots \\ x_m(t_{m-1}) \\ \vdots \\ (t_m - t_{m-1})^{k-1} x_m^{(k-1)}(t_{m-1}) \\ x_m(t_m) \\ (t_m - t_{m-1})^{k-1} x_m^{(k-1)}(t_m) \end{bmatrix}
\end{aligned}$$

Like before, we just omit rows where a condition is unconstrained. Also, note that except for constraints at  $t_0$  and  $t_m$ , every other constraint must be include twice. The equation for  $A[i, j](t)$  is the same of (4.10)

We must also take into account for constraints that ensure that when the trajectory switches from one piece to another at the sepoints, position and all the derivative lower than  $k$  remain continuous, for a smooth path. In other words, is require that

$$A_{cont}\mathbf{c} = \mathbf{b}_{cont} \quad (4.18)$$

$$\begin{bmatrix} x_1(t_1) - x_2(t_2) \\ \vdots \\ x_1^{(k-1)}(t_1) - x_2^{(k-1)}(t_1) \\ \vdots \\ x_{m-1}(t_{m-1}) - x_m(t_{m-1}) \\ \vdots \\ x_{m-1}^{(K-1)}(t_{m-1}) - x_m^{(K-1)}(t_{m-1}) \end{bmatrix} = 0$$

Translating to the non-dimeensionalized case,  $\tau_0 = 0$ ,  $\tau_1 = 1$ , and

$$A_{cont}\mathbf{c} = \mathbf{b}_{cont} \quad (4.19)$$

$$\begin{bmatrix} x_1(\tau_1) - x_2(\tau_2) \\ \vdots \\ \frac{1}{(t_1-t_0)^{k-1}}x_1^{(k-1)}(\tau_1) - \frac{1}{(t_2-t_1)^{k-1}}x_2^{(k-1)}(\tau_1) \\ \vdots \\ x_{m-1}(\tau_1) - x_m(\tau_0) \\ \vdots \\ \frac{1}{(t_{m-2}-t_{m-1})^{k-1}}x_{m-1}^{(K-1)}(\tau_1) - \frac{1}{(t_m-t_{m-1})^{k-1}}x_m^{(K-1)}(\tau_0) \end{bmatrix} = 0$$

and then

$$\begin{bmatrix} A_{cont}(t_1) & 0 & \dots & 0 \\ 0 & A_{cont}(t_2) & \dots & 0 \\ \vdots & 0 & \dots & 0 \\ 0 & \dots & 0 & A_{cont}(t_{m-1}) \end{bmatrix} \mathbf{c} = 0 \quad (4.20)$$

where

$$A_{cont}[i, j](t_z) = \begin{cases} \frac{1}{(t_z-t_{z-1})^i} \prod_{z=0}^{i-1} (n-z-j) \tau_1^{n-j-i}, & n-j \geq i \wedge j \leq n \\ 0, & n-j < i \wedge j \leq n \\ -\frac{1}{(t_{z+1}-t_z)^i} \prod_{z=0}^{i-1} (1-z-j) \tau_0^{1-j-i}, & 1-j \geq i \wedge j > n \\ 0, & 1-j < i \wedge j > n \end{cases} \quad (4.21)$$

$$i = 0, \dots, k-1, \quad j = 0, \dots, 2(n+1)$$

The final constraints  $A\mathbf{c} = \mathbf{b}$  take then the final form

$$\begin{aligned} A\mathbf{c} &= \mathbf{b} \\ \begin{bmatrix} A_{\text{endpoint}} \\ A_{\text{cont}} \end{bmatrix} \mathbf{c} &= \begin{bmatrix} b_{\text{endpoint}} \\ 0 \end{bmatrix} \end{aligned} \quad (4.22)$$

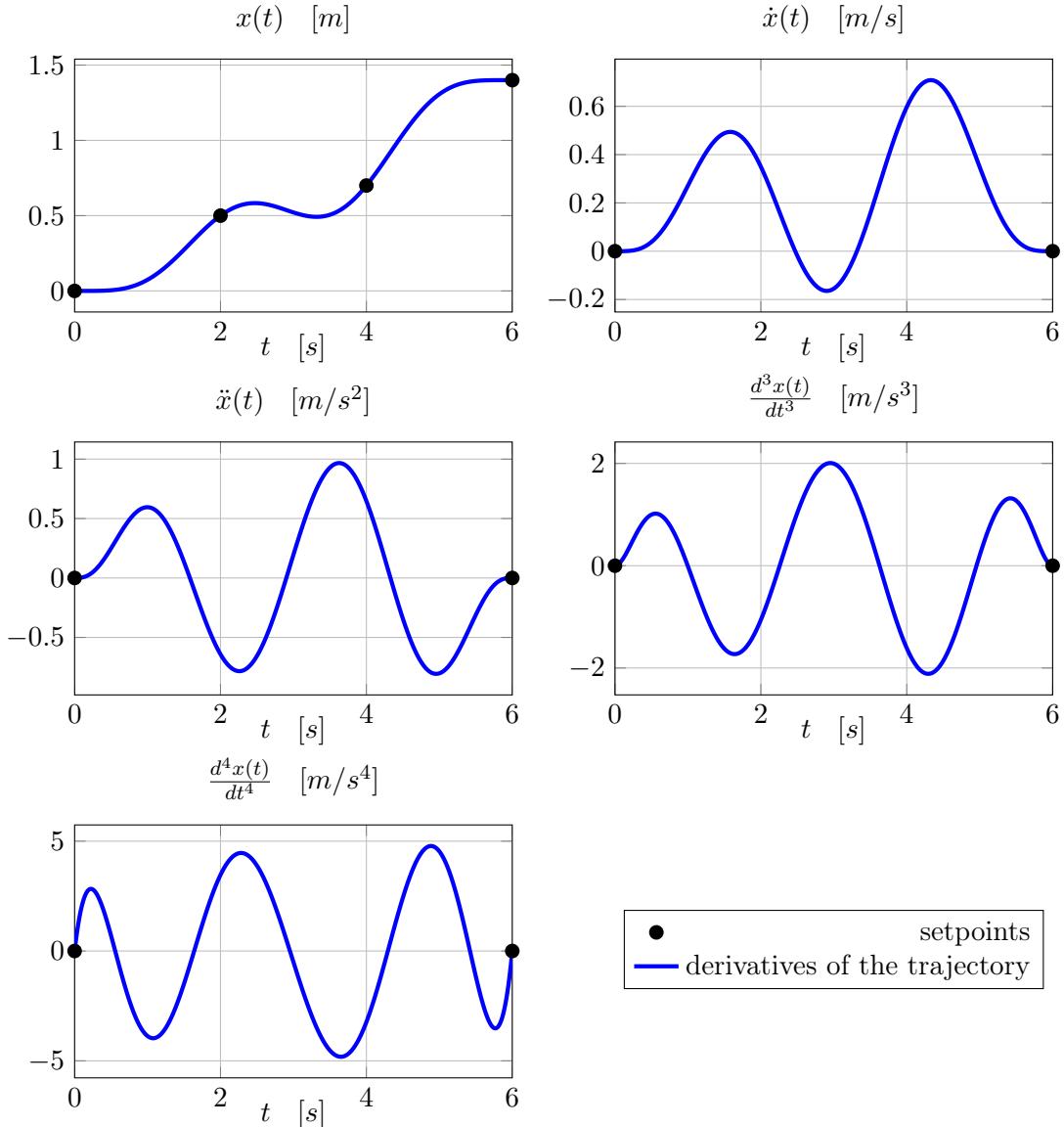


Figure 4.1: Generated trajectory with its derivatives.

In figure 4.1 is reported the first dimension of a generated trajectory evolving over the time. In particular in blue are plotted the trajectory and its first four derivatives, till the snap. Instead in black are plotted the setpoints for the trajectory and the initial and final conditions for each derivative, that are equal to zero. Notice that between setpoints two and three, the trajectory is not as expected, in the sense that it first tends to go far from the desired setpoint and then reach it. That's because this is only one dimension of a more complex three dimension trajectory.

## 4.4 Adding corridor constraints

In this section, corridors constraints will be added in the cost function (4.3). For corridor constraints, we mean that the desire trajectory must be inside a corridor, this why for a safe obstacle avoidance and navigation algorithm, the vehicle must respect distance between walls and obstacles. To do this, we first define  $\mathbf{u}_i$  as the unit vector along the segment from setpoint  $\mathbf{r}_i$  and setpoint  $\mathbf{r}_{i+1}$ .

$$\mathbf{u}_i = \frac{\mathbf{r}_{i+1} - \mathbf{r}_i}{\|\mathbf{r}_{i+1} - \mathbf{r}_i\|} \quad (4.23)$$

The perpendicular distance vector,  $\mathbf{d}_i(t)$ , from segment  $i$  is defined as

$$\mathbf{d}_i(t) = (\mathbf{r}_d(t) - \mathbf{r}_i) - ((\mathbf{r}_d(t) - \mathbf{r}_i) \cdot \mathbf{t}_i) \mathbf{u}_i \quad (4.24)$$

where  $\mathbf{r}_d(t)$  is the desire trajectory at instant  $t$ . A corridor width on the infinity norm,  $\delta_i$ , is defined for each corridor as follow

$$\|\mathbf{d}_i(t)\|_\infty \leq \delta_i \quad \text{while } t_i \leq t \leq t_{i+1} \quad (4.25)$$

The reason to write the constraint like that, is because it can be incorporate into the QP problem by introducing  $n_c$  intermediate points as

$$\left| x_a \cdot \mathbf{d}_i \left( t_i + \frac{j}{1+n_c} (t_{i+1} - t_i) \right) \right| \leq \delta_i \quad \text{for } j = 1, \dots, n_c \quad (4.26)$$

where for  $x_a$  we mean that this procedure must be compute for  $x_W$ ,  $y_W$  and  $z_W$ , with  $\mathbf{x}_W = [x_W \ y_W \ z_W]^T$ . Of course a corridor constraint in the desire yaw doesn't have sense. To do this, we introduced inequality constraints of the form  $A_{ineq}\mathbf{c} \leq \mathbf{b}_{ineq}$ . To find  $A_{ineq}$ , we first break down the inequality (4.26) into

$$(x_a \cdot \mathbf{d}_i(t_i + \frac{j}{1+n_c} (t_{i+1} - t_i))) \leq \delta_i \quad (4.27)$$

$$-(x_a \cdot \mathbf{d}_i(t_i + \frac{j}{1+n_c} (t_{i+1} - t_i))) \leq \delta_i \quad (4.28)$$

This result in a total of  $2 \cdot 2 \cdot n_c$  constraints for each corridor constraint. Then by performing some math, the matrix  $A_{ineq}$  and the vector  $b_{ineq}$  can be deduce <sup>1</sup>.

In figure 4.2 are reported examples of trajectories, with and without corridor constraints. For better understanding are reposrt 2D trajectories, but the same example could be made also for 3D trajectories. The corridor constraint is present only between two setpoints, setpoint 2 and setpoint 3. As is possible to see, the trajectory remain in beetween the constraint, but become less smooth in compare to the one without constraints. Moreover, notice that the entire trajectory is different and not only the segment in between the corridor, this is another important advantage of using this technique to generate trajectores.

<sup>1</sup>The formulation of such matrix is very eavy and difficult to understand, that's why is not report. Or maybe put in a appendix

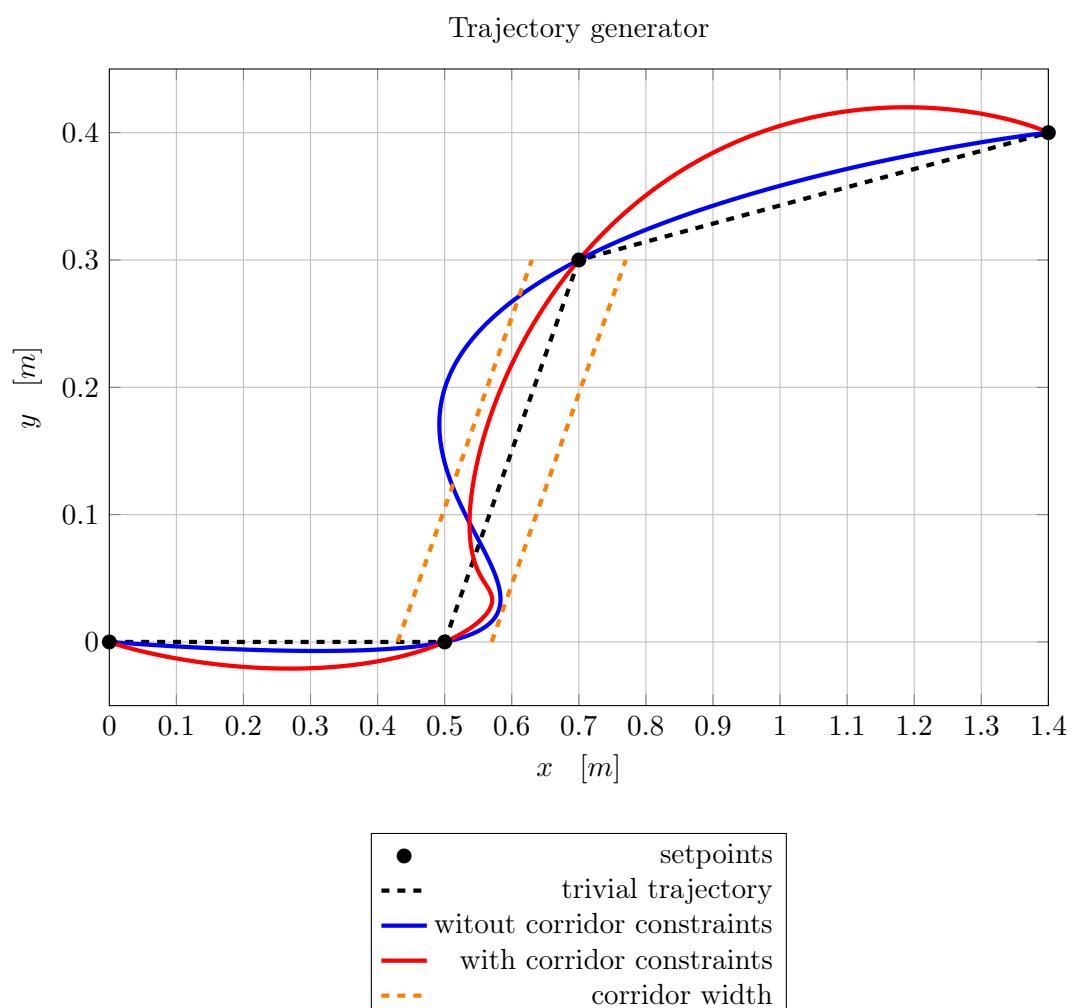


Figure 4.2: Setpoints and desire trajectory, with and without corridor constraints between setpoints 2 and 3.

# 5

## Control

The last important step of this work, is to derive a control law, able to track the desire trajectory and to compensate for the movement of the sensor.

### 5.1 Position tracking controller in $SO(3)$

In this section, will be introduce a tracking control for the vehicle, based on the work [18]. We will use this control because it has been show to work well in many different applications, from precision flights, to fast and aggressiv flights. In particular, starting from the model of the UAV, we will derive a position tracking control, based on  $SO(3)$  group.

The *general linear group* of order 3,  $GL(3)$ , is a algebraic group composed by all the non singular matrices  $A \in \mathbb{R}^{3 \times 3}$  with matrix product.

The *orthogonal group* of order 3,  $O(3)$ , is deefine as

$$O(3) = \{A \in GL(3) : A^T A = I\} \subseteq GL(3)$$

Is easy to prove that if  $A$  belongs to  $O(3)$ , then  $\det[A] = \pm 1$ .

The *special orthogonal grouop* of order 3,  $SO(3)$ , is define as

$$SO(3) = \{A \in O(3) : \det[A] = 1\}$$

Belong to this gruop the rotation matrices, and is because the controller will use errors based ion matrices, we need to keep all the results in the group  $SO(3)$ .

To derive the control law, we first need to define the tracking errors. In particular the position and the velocity tracking errors are given by, respectively

$$\mathbf{e}_x = \mathbf{x} - \mathbf{x}_d \tag{5.1}$$

$$\mathbf{e}_v = \dot{\mathbf{x}} - \dot{\mathbf{x}}_d \tag{5.2}$$

where the subscript  $d$  stands for desire. The attitude error is instead defined as

$$\mathbf{e}_R = \frac{1}{2} (R_c^T R - R^T R_c)^\vee \quad (5.3)$$

where  $R$  is the rotation matrix that encodes the actual attitude of the UAV, while  $R_c(t) \in SO(3)$  is the computed attitude matrix, that must belong to the special orthogonal group. In fact we can define it as

$$\mathbf{b}_{1,c} = [\cos(\psi_d) \quad \sin(\psi_d) \quad 0]^T \quad (5.4)$$

$$\mathbf{b}_{3,c} = -\frac{k_x \mathbf{e}_x + k_v \mathbf{e}_v - g \mathbf{e}_3 - \ddot{\mathbf{x}}_d}{||k_x \mathbf{e}_x + k_v \mathbf{e}_v - g \mathbf{e}_3 - \ddot{\mathbf{x}}_d||} \quad (5.5)$$

$$\mathbf{b}_{2,c} = \frac{\mathbf{b}_{3,c} \times \mathbf{b}_{1,c}}{||\mathbf{b}_{3,c} \times \mathbf{b}_{1,c}||} \quad (5.6)$$

$$R_c = [ \mathbf{b}_{2,c} \times \mathbf{b}_{3,c} \mid \mathbf{b}_{2,c} \mid \mathbf{b}_{3,c} ] \quad (5.7)$$

Where  $\psi_d$  is the desire yaw,  $\mathbf{e}_3$  is the third dimension canonical vector,  $k_x$  and  $k_v$  are positive control constants. The *vee* map  $\cdot^\vee$  is the inverse of the *hat* map  $\hat{\cdot}: \mathbb{R}^3 \rightarrow SO(3)$  defined as

$$\begin{aligned} \mathbf{v} &= [v_1 \quad v_2 \quad v_3]^T \\ \hat{\mathbf{v}} &= \begin{bmatrix} 0 & -v_3 & v_2 \\ v_3 & 0 & -v_1 \\ -v_2 & v_1 & 0 \end{bmatrix} \end{aligned} \quad (5.8)$$

The angular velocity error  $e_\omega$  depends only from the desire yaw, since our trajectory generator computes the desire yaw till the second derivative. However, it can be computed simply by

$$\mathbf{e}_\omega = \boldsymbol{\omega} - R^T R_C \hat{\boldsymbol{\omega}}_c \quad (5.9)$$

where  $\hat{\boldsymbol{\omega}}_c = R_c^T \dot{R}_c$ .

The final control law is then divided into two contributions, one for the total force and one for the torque

$$f = -(k_x \mathbf{e}_x + k_v \mathbf{e}_v - g \mathbf{e}_3 - \ddot{\mathbf{x}}_d)^T R \mathbf{e}_3 \quad (5.10)$$

$$\tau = -k_R \mathbf{e}_R - k_\omega \mathbf{e}_\omega + \boldsymbol{\omega} \times I_{cm} \boldsymbol{\omega} \quad (5.11)$$

where  $k_R$  and  $k_\omega$  are again positive control gains. As is possible to see, this controller is very simple to implement and can be proved that the control is exponentially stable, if the initial conditions are sufficiently small<sup>1</sup>. Of course the control law is not complete, because what we can control are the inputs to the motors and not the force and torques. So to compute the linear acceleration in the body frame we just need to do

$$\ddot{\mathbf{x}}_B = \frac{1}{m} \cdot [0 \quad 0 \quad f]^T \quad (5.12)$$

---

<sup>1</sup>see the paper [18] for more details and the proof

For the angular acceleration in the body frame, in theory we just need to compute

$$\dot{\omega}_B = I_{cm}^{-1} \tau \quad (5.13)$$

Then, by using the results from section 2.2.2, we can do the computation of the matrix that encode the law from motors' inputs to linear and angular acceleration and obtain

$$\begin{bmatrix} \vdots \\ u_i^2 \\ \vdots \end{bmatrix} = \underbrace{\left[ \dots \begin{array}{c} A_F \\ 0 \\ 0 \\ A_F \end{array} + \begin{array}{c} 0 \\ 0 \\ 0 \\ B_F \end{array} \dots \right]^{-1}}_{T^{-1}(\beta)} \cdot \begin{bmatrix} f \\ \tau \end{bmatrix} \quad (5.14)$$

where the matrix  $T^{-1}(\beta)$  is the invers of the estimated parameters, except that is encoded only one row for the force, since three propellers are parallel to the frame of the quadrotor. However the problem is in the computation of the inverse of the inertia matrix, since the estimated parameters are coupled with the component of the inertia and is not possible to estimate directly  $A_F$  and  $B_F$ . A naive approach is simply use the inertia computed with the CAD model. Of course, this approach will add errors, since the CAD doesn't provide a perfect data. However, this inverse will be multiplied with  $k_R$  and  $k_\omega$  and then just a simple retuning of the parameters will be necessary. Instead, the term  $I_{cm}^{-1}(\omega \times I_{cm}\omega)$  as said in section 3.1 is very small and will not introduce significant errors.

### 5.1.1 Adding the rotating platform

The previous controller was derived without the rotating platform. To introduce the compensation for the movement of the cart, first of all we introduce the compensation for the moving COG. To do this we simply modify the terms  $\mathbf{l}_i$  in the matrix  $T(\beta)$  and then compute the inverse at every iteration. Of course, to do this, we need to know precisely the position  $\gamma$  of the cart, this can be done since the motor that drive the cart is provided of encoder [3]. The changes in the inertia matrix are instead computed with the CAD, a more precisely solution could be to compute the system identification with the moving cart or, if it doesn't work, compute the system identification multiple times with different positions of the sensors and then interpretate the system as a piecewise system.

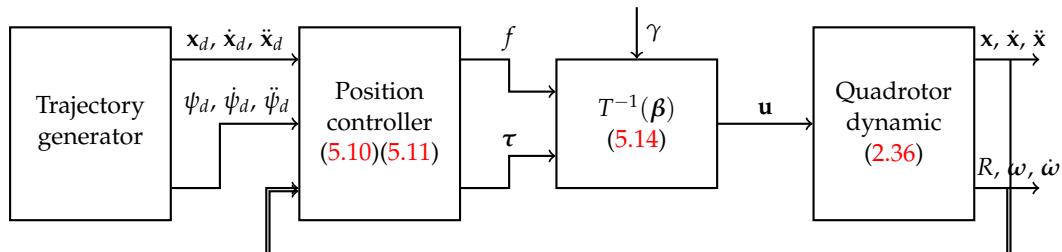


Figure 5.1: Block diagram of the control scheme (the subscripts that indicate the appartenence of the frame are omitted).

### 5.1.2 Simulation Results

In figure 5.2 is reported a simple example of trajectory tracking. In particular, the desire trajectory is a step for the desire height  $z$  and hovering for the remaining variables. Moreover is illustrate why the cart compensation is important.

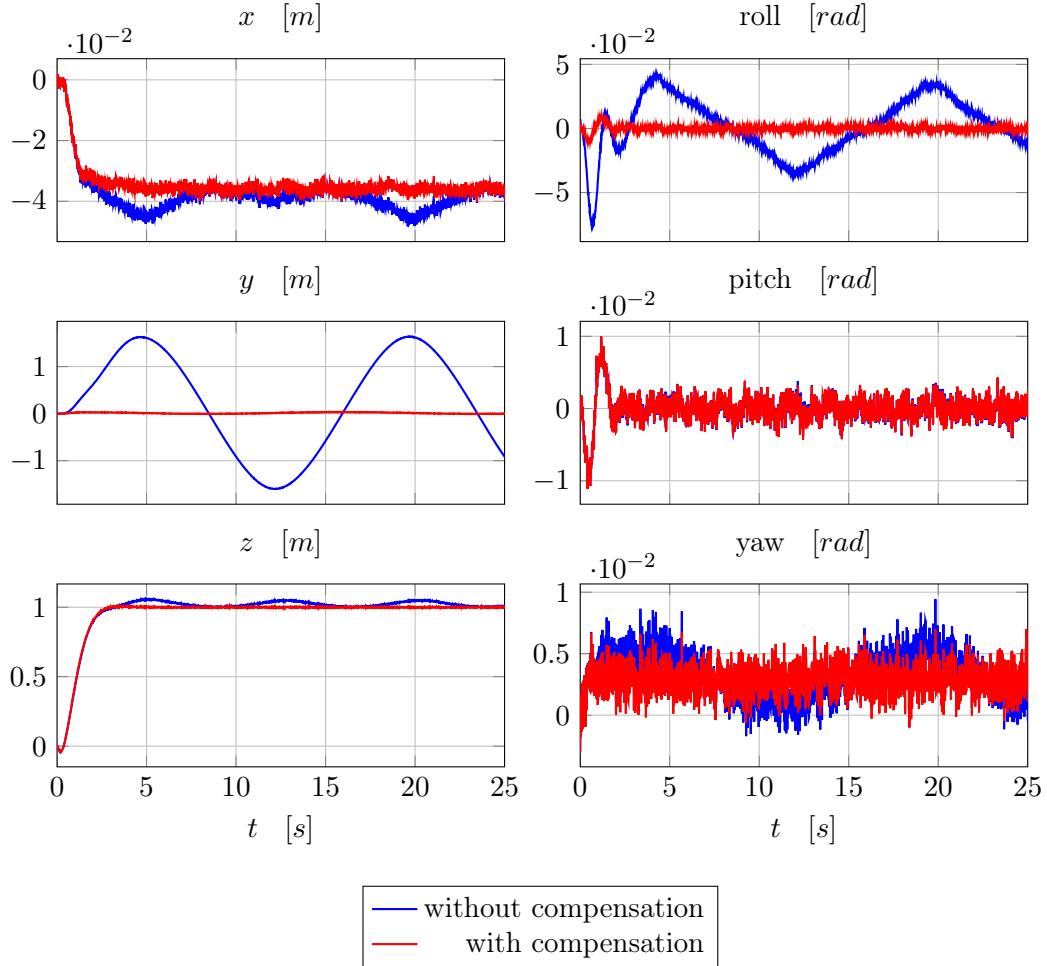


Figure 5.2: Simulation results to a step input in the desire  $z$ , then hovering. Without and with the cart compensation. The mass of the moving cart is one third of the real mass. The cart is moving with a constant speed of 6 rpm, the maximum possible for the engine adopted [3].

In red are depicted the red trajectories with the compensation, while, in blue, without the compensation. Note how the results is much more stable with the compensation. Note also that in the simulation the mass of the moving cart is one third of the real mass, this because if we simulate without compensation, the controller is not able to keep the vehicle balance, resulting in a complete loss of control.

In figure 5.3 is instead depicted a simulation with the full mass of the cart. The control algorithm is the one with the cart compensation and the desire trajectory is again a step input for the desire height and constant for the other inputs. As is possible to see the control works quite well but some oscillations are again noticeable. Those oscillations are come from the engines. It means that we know the position of the cart at the actual time and than apply the control compensation. However the engines

apply a delay to the system and this compensation is delayed by the time constant of the motors.

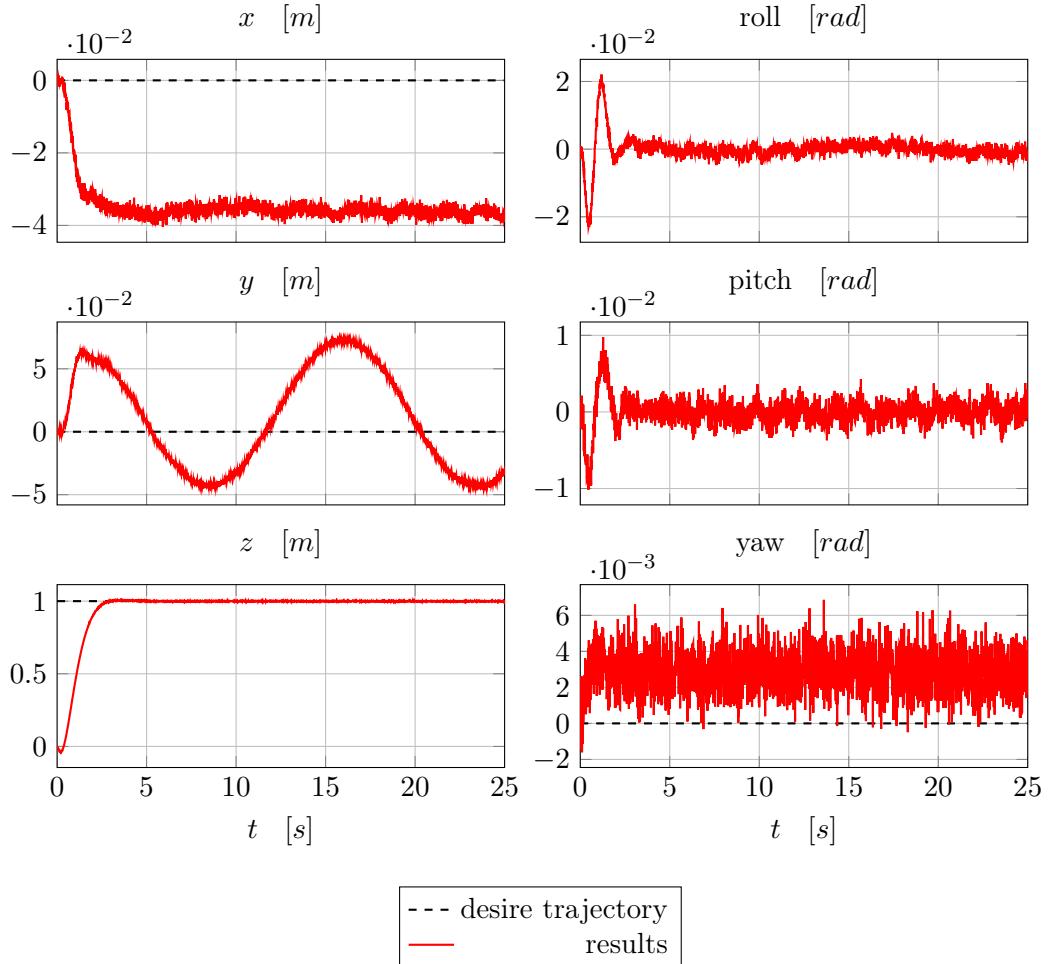


Figure 5.3: Simulation results to a step input in the desire  $z$ , then hovering. The cart is moving with a constant speed of 6 rpm, the maximum possible for the engine adopted [3].

This delay is very clear especially in the  $y$  position, since the moving part is in the  $y$  and  $z$  axis. A solution for this problem could be to estimate the position of the cart in advance with an accurate model of this. Others problems are some bias errors in the steady state, however these problems come from the simulated bias of the sensors. These errors are however of the order of the millimeter, and no solutions are provided in this thesis, but is a direction for a future work in this project.



# 6

## Conclusions and future works

write te conclusions



## Bibliography

- [1] **P. Pounds, R. Mahony, and P. Corke.** Modelling and control of a large quadrotor robot. *Control Engineering Practice* 18, 2010. [1](#)
- [2] **C. Mary, L. C. Totu, and S. Konge Koldbæk.** Modelling and Control of Autonomous Quad-Rotor. *Aalborg Universitet, June 2010.* [1](#)
- [3] **C. Navarro Leoncio.** Design of the Prometheus UAV. *Master thesis project, Luleå university of technology*, 2016. [2](#), [13](#), [37](#), [38](#), [39](#)
- [4] **J. Solà.** Quaternion kinematics for the error-state KF. *February 2, 2016.* [7](#)
- [5] **E. Fresk, and G. Nikolakopoulos.** Full Quaternion Based Attitude Control for a Quadrotor. *2013 European Control Conference (ECC), July.* [8](#)
- [6] **J. Diebel.** Representing Attitude: Euler Angles, Unit Quaternions, and Rotation Vectors. *Stanford University*, 2006. [8](#)
- [7] **T. Bresciani.** Modelling, Identification and Control of a Quadrotor Helicopter. *Department of Automatic Control, Lund University, October 2008.* [10](#)
- [8] **E. Fresk, and G. Nikolakopoulos.** Experimental Model Derivation and Control of a Variable Pitch Propeller Quadrotor. *IEEE Multi-Conference on System and Control, 2014.* [11](#)
- [9] **E. Fresk, R. J. Cotton, and G. Nikolakopoulos.** Generalized Model Identification for Multirotors: Towards applications in Auto Tuning. 2016. [11](#), [17](#), [18](#)
- [10] **F. P. Beer, E. R. Johnston, D. F. Mazurek, P. J. Cornwell, and B. P. Self.** Vector Mechanics for Engineers, Statics and Dynamics. *Tenth edition.* [13](#)
- [11] **N. Abas, A. Legowo, and R. Akmeliauwati.** Parameter Identification of an Autonomous Quadrotor. *2011 4th International Conference on Mechatronics, 17-19 May 2011, Kuala Lumpur, Malaysia.* [19](#)
- [12] **G. Welch, and G. Bishop.** An Introduction to the Kalman Filter. *University of North Carolina at Chapel Hill, Department of Computer Science.* [19](#), [20](#)
- [13] **J. Barraquand, B. Langlois, and J.-C. Latombe.** Numerical Potential Field Techniques for Robot Path Planning. *Stanford University, California*, 1989. [25](#)
- [14] **M. Blösch, S. Weiss, D. Scaramuzza, and R. Siegwart.** Vision Based MAV Navigation in Unknown and Unstructured Environments. *Autonomous Systems Lab, ETH Zurich.* [25](#)

- [15] **D. Mellinger, and V. Kumar.** Minimum Snap Trajectory Generation and Control for Quadrotors. *IEEE International Conference on Robotics and Automation, Shanghai, China, 2011.* [25](#)
- [16] **C. Richter, A. Bry, and N. Roy.** Polynomial Trajectory Planning for Quadrotor Flight. *Department of Aeronautics and Astronautics, Massachusetts Institute of Technology, Cambridge.* [25](#)
- [17] **D. W. Mellinger.** Trajectory Generation and Control for Quadrotors. *University of Pennsylvania, 2012.* [26](#)
- [18] **T. Lee, M. Leok, and N. H. McClamroch.** Nonlinear Robust Tracking Control of a Quadrotor UAV on  $SE(3)$ . *2012 American Control Conference, Fairmont Queen Elizabeth, Montréal, Canada, June 27 - June 29, 2012.* [35](#), [36](#)