1\_Trimmed\_VaAlphaBeta\_bodyframe/

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| **Case Folder/CASE No.** | **Common Parameter** | **Case Conditions** |
| 1\_Trimmed\_VaAlphaBeta\_bodyframe/results/ **case 1** |  | In **vtol\_parameters.m**  vtol.Va0 = 10;  gamma = 5\*pi/180;  R = Inf; |
| 1\_Trimmed\_VaAlphaBeta\_bodyframe/results/ **case 2** |  | In **vtol\_parameters.m**  vtol.Va0 = 17;  **gamma = 5\*pi/180;**  R = Inf; |
| 1\_Trimmed\_VaAlphaBeta\_bodyframe/results/ **case 3** |  | In **vtol\_parameters.m**  **vtol.Va0 = 10;**  **gamma = 0\*pi/180;**  R = Inf; |
| 1\_Trimmed\_VaAlphaBeta\_bodyframe/results/ **case 4** |  | In **vtol\_parameters.m**  **vtol.Va0 = 40;**  **gamma = 0\*pi/180;**  R = Inf; |
| 1\_Trimmed\_VaAlphaBeta\_bodyframe/results/ **case 5** |  | In **vtol\_parameters.m**  **vtol.Va0 = 15;**  gamma = 0\*pi/180;  R = Inf; |
| 1\_Trimmed\_VaAlphaBeta\_bodyframe/results/ **case 6** |  | * With Propulsion equations from Errata.      * Aerodynamic coefficients: AppendixE * Mass = 11 kg   vtol.Va0 = 10;  gamma = 5\*pi/180;  R = Inf; |
| 1\_Trimmed\_VaAlphaBeta\_bodyframe/results/ **case 7** |  | * With Propulsion equations from Errata like in case 6. * Aerodynamic coefficients: Errata coefficients * Mass 11kg   vtol.Va0 = 5;  gamma = 0\*pi/180;  R = Inf; |
| 1\_Trimmed\_VaAlphaBeta\_bodyframe/results/ **case 8** |  | * With Propulsion equations from Errata like in case 6. * Aerodynamic coefficients: **Appendix E** * Mass 11kg   vtol.Va0 = 10;  gamma = 0\*pi/180; % desired flight path angle (radians)   * R = Inf; |
| 1\_Trimmed\_VaAlphaBeta\_bodyframe/results/ **case 9** |  | * With Propulsion equations from Errata like in case 6. * Aerodynamic coefficients: **Errata coefficients (Linearization coeff is too large!)**      * Mass 11kg   vtol.Va0 = 10;  gamma = 0\*pi/180;  R = Inf; |
| 1\_Trimmed\_VaAlphaBeta\_bodyframe/results/ **case 10** |  | * With Propulsion equations from Errata like in case 6. * Aerodynamic coefficients: **Appendix E** * Mass 11kg   **vtol.Va0 = 17;**  gamma = 0\*pi/180;  R = Inf; |
| 1\_Trimmed\_VaAlphaBeta\_bodyframe/results/ **case 11** |  | * With Propulsion equations from Errata like in case 6. * Aerodynamic coefficients: **Appendix E** * Mass 11kg   **vtol.Va0 = 10;**  gamma = 5\*pi/180; % desired flight path angle (radians)   * R = 50; |
| 1\_Trimmed\_VaAlphaBeta\_bodyframe/results/ **case 12** |  | * With Propulsion equations from Errata like in case 6. * Aerodynamic coefficients: **Appendix E** * Mass 11kg   **vtol.Va0 = 17;**  gamma = 5\*pi/180  R = 50; |