



US Department
of Transportation
Federal Aviation
Administration

**MAJOR REPAIR AND ALTERATION
(Airframe, Powerplant, Propeller, or Appliance)**

Form Approved
OMB No. 2120-0020
2/28/2011

Electronic Tracking Number

For FAA Use Only

INSTRUCTIONS: Print or type all entries. See Title 14 CFR §43.9, Part 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. §44701). Failure to report can result in a civil penalty for each such violation. (49 U.S.C. §46301(a))

1. Aircraft	Nationality and Registration Mark N959JT	Serial No. T20608983		
	Make CESSNA	Model T206H	Series	
2. Owner	Name (As shown on registration certificate) PSL Surveys	Address (As shown on registration certificate) Address PO Box # 756 City Bristow State VA Zip 20136 Country United States		

3. For FAA Use Only

4. Type		5. Unit Identification			
Repair	Alteration	Unit	Make	Model	Serial No.
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in Item 1 above)	_____
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT	_____	_____	_____
<input type="checkbox"/>	<input checked="" type="checkbox"/>	PROPELLER	_____	_____	_____
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type	_____	_____
			Manufacturer		

6. Conformity Statement

A. Agency's Name and Address		B. Kind of Agency	
Name Straight Flight, Inc.	Address 13251 E. Control Tower Road, K12	U. S. Certificated Mechanic	Manufacturer
City Englewood	State Colorado	Foreign Certificated Mechanic	C. Certificate No.
Zip 80112	Country United States	Certificated Repair Station	OMKR399L
		Certificated Maintenance Organization	

D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

Extended range fuel per 14 CFR Part 43 App. B	<input type="checkbox"/>	Signature/Date of Authorized Individual Ernest L. Smith IV	November 13, 2013
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7. Approval for Return to Service

Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is Approved Rejected

BY	FAA Fit. Standards Inspector		Manufacturer	Maintenance Organization	Persons Approved by Canadian Department of Transport
	FAA Designee	<input checked="" type="checkbox"/>	Repair Station	Inspection Authorization	Other (Specify)
Certificate or Designation No. OMKR399L		Signature/Date of Authorized Individual Ernest L. Smith IV			

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N959JT

November 13, 2013

Nationality and Registration Mark

Date

Installed Gomolzig Flugzeug und Maschinenbau STC #SA03134NY "Low Noise Silencer Exhaust Extension" IAW Installation Instructions # II_Eng_16291_rev2_20120301 dated 1-03-2013 and Drawing #CT206H-606570.

Instructions for Continued Airworthiness: The Gomolzig Maintenance Manual Supplement issue 15.5.2012 was inserted into the aircraft records which requires 100 hour and 500 hour separate inspections.

Gomolzig Flight Manual Supplement issue 08.07 was inserted into the aircraft flight manual.

Weight and Balance was amended and inserted into the aircraft flight manual.

Information on this alteration is on file under Straight Flight's project number 7137.

*****END*****



Additional Sheets Are Attached



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1. Aircraft	Nationality and Registration Mark N959JT	Serial No. T20608983	
	Make Cessna	Model T206	Series
2. Owner	Name (As shown on registration certificate) PSL Surveys	Address (As shown on registration certificate) Address PO Box 756 City Bristol Zip 20136	State VA Country USA

3. For FAA Use Only

The technical data identified herein has been found to comply with the applicable airworthiness requirements and is hereby approved for use only on the above described aircraft, subject to conformity inspection by a person authorized in CFR title 14, Part 43, section 43.7.

Approving Inspector: Julie A. Sizemore Date: 3/29/2013
Denver FSDO, NM-03

4. Type		5. Unit Identification			
Repair	Alteration	Unit	Make	Model	Serial No.
<input type="checkbox"/>	<input type="checkbox"/>	AIRFRAME	_____	(As described in Item 1 above)	_____
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT	_____	_____	_____
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER	_____	_____	_____
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type Manufacturer	_____	_____

6. Conformity Statement

A. Agency's Name and Address		B. Kind of Agency		
Name	Philip Glasgow	<input checked="" type="checkbox"/> U. S. Certificated Mechanic	Manufacturer	
Address	2533 Dallas Creek Court	<input type="checkbox"/> Foreign Certificated Mechanic	C. Certificate No.	
City	Fort Collins	<input type="checkbox"/> Certificated Repair Station		
Zip	80528	<input type="checkbox"/> Certificated Maintenance Organization	A&P 3292572 IA	

D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

Extended range fuel per 14 CFR Part 43 App. B	<input type="checkbox"/>	Signature/Date of Authorized Individual <u>Philip Glasgow</u> 4/2/13
---	--------------------------	---

7. Approval for Return to Service

Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is Approved Rejected

BY	FAA Flt. Standards Inspector	Manufacturer	Maintenance Organization	Persons Approved by Canadian Department of Transport
	FAA Designee	Repair Station	<input checked="" type="checkbox"/> Inspection Authorization	Other (Specify)
Certificate or Designation No. A&P 3292572 IA		Signature/Date of Authorized Individual <u>Philip Glasgow</u> 4/2/13		

NOTICE

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8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N959JT

4/2/13

Nationality and Registration Mark

Date

-Installed a Paravion Technology Inc Infared camera mounting provisions IAW STC STC SA 00295DE for a L3 Wescam MX10 camera system.

-Installed a Churchill Augmented Reality System IAW manufacturers installation drawings # ARS 500C-201210 Rev 6 10/24/12. Power is supplied from the avionics buss and is protected using a Klixon C/B P/N 7277-2-3 labeled "Mapping" Mounted the ARS system to the above installed Infrared camera mount and secured the ARS to the mount by fabricating 2 X support brackets. Ref Paravion Technology drawing 206ARS-1000 sheet 1 for full fabrication details. Mounted the GPS antenna to the roof of the aircraft structure at station 104.0 using manufacturer provided hardware.

-Mounted a 9.0"Airborne display monitor into the instrument panel. The primary display monitor is mounted to the instrument panel on the R/H side using 2X MS24693-363 screws. Attached 2X MS21059-L3 nut plates to the instrument panel using 4 X MS21426-3-4 countersunk rivets. The remote control unit is provided power from the avionics buss and is protected using a 3 Amp Klixon C/B P/N 7277-2-3. And is labeled "Monitor". The remote control unit for the monitor is mounted to aircraft structure behind the instrument panel. Attached the control unit to the support brackets using 4 X screws P/N MS24693S26 and 4 X clip nuts P/N 294667. Fabricated the two supports from stock 6061 T6 aluminium and machined the support brackets. Ref Paravion Technology drawing 206ARS-1000 sheet 5 for fabrication details. Attached the support brackets to the instrument panel using 4 X screws P/N and 4 X nutplates P/N which are riveted to the support brackets using 8 X MS20426AD3-5 rivets. Fabricated a support brace from 6061 T6 aluminium 0.063" and bent to a 90 deg angle. Attached the brace to the supports using 2 X nutplates P/N MS21059L06 and 2 X screws P/N MS24693S26. Fabricated a plate for the remote control controls from 6061 T6 aluminum 1.5" X 4". Secured the power switch, dimmer switch, menu control switch, Video selection switch & the Downlink switch to this panel using the manufacturers provided switches. Secured the panel to the arm rest of the interior plastic using 4 X MS35206-226 screws, 4 X AN 960-6L washers & 4 X MS21083N06 nuts.

-Mounted The Jantec Downlink Control ECU to the floor at station 133.75 using 4 X MS27039-1-09 screws. Attached 3 X nut plates P/N MS2105L3N and attached to the existing structure using 6 X CR3213 4-4 rivets. Fabricated a doubler from 6061 T6 .063" 8.5" X .7 X.7 angle. Attached two of the afore mentioned nut plates to this doubler using the rivets mentioned above. Installed the Jantec Down link IAW manufacturers Dwgs . System is protected using a Klixon C/B switch P/N 7270-3-10 and is labeled "Down Link" Mounted two antennas on the bottom of the aircraft. Mounted the first antenna at station 150.0" on the bottom of the aircraft to the R/H side of the aircraft center line. Fabricated a doubler from 6061 T6 aluminium 4" X 5". Attached the antenna to the aircraft using 4 X P/N MS51987-48 screws, 4 X P/N AN960C8 washers & 4 X P/N MS21042-L08 nuts. Mounted the second antenna to the bottom of the aircraft at station 159.0" to the L/H side of the center line. Fabricated a doubler from 6061 T6 aluminium 4" X 5". Attached the antenna to the aircraft using 4 X P/N MS51987-48 screws, 4 X AN960C8 P/N washers & 4 X P/N P/N MS21042-L08 screws. Mounted the control head to the center console using 4 X P/N 2-56 screws. Fabricated a double and machined to fit. ref Paravion Technology Dwg C206ARS-1000 sheet 11 for fabrication details.

-Installed 2 X Aux Foot switches on the floor at station location 20.00". Fabricated foot switch holder form the same material as mentioned above for the center console and installed a 2 X switches P/N M8805/55-001 X 2. Attached the Foot switch housing Using 2 x MS35206-228 screws and 2 X AN960JD6L washers. to the floor using 3 X Nut plates P/ N MS21075L06 & 1 X MS21069L06 nut plate. Attached the nut plates to the floor using 8 X MS20426AD3-3.5 Rivets.

Additional Sheets Are Attached

NOTICE

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8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N959JT

Nationality and Registration Mark

4/2/13

Date

-Fabricated a breaker panel from .25" X 6.5" X 6.5" stock. Machined the required holes and location for the various connectors required. fabricated 2 X angles form .040" 6.5"X .75" X 1.25" 6061 T6 aluminium. Attached the angles to the breaker panel using 6 X screws P/N MS24693S26. Attached the assembly to the airframe using 4 X screws P/N MS24693S26. Attached 4 X nut plates P/N MS21069L06 to the existing structure using 8 X MS20426AD-3-4 rivets. Ref Paravion Technology Dwg C206ARS-1000 sheet 8 for fabrication details.

- Fabricated a carbon fiber housing to mount 2 X USB ports and 1 X hand controller cannon plug. Attached the housing to the aircraft structure on the floor between the seats at station 55.0 just aft of the existing vent using 4 X MS21075L3N nut plates. Attached the nut plates to the the floor using 4 X MS20426-3-4 countersunk rivets. used 4 X screws P/N MS27039-1-09 screws and 4 X AN960C10L washers. Ref attached Paravion Technology dwg C206ARS-1000 sheet 10 for fabrication details of the housing.

- Fabricated a mount for the existing Motorola XTVA radio housing form .063 6061 T6 aluminium 9" X 2". Attached the mount to the floor aft of the USB housing at station 65.0 using 2 X MS21075L3N nut plates. Attached the nut plates to the the floor using 4 X MS20426-3-4 countersunk rivets. used 2 X screws P/N MS27039-1-09 screws and 2 X AN960C10L washers. Ref Paravion Technology drawing C206ARS-1000 sheet 9 for fabrication details.

Wire gauge selection was done in accordance with AC43-13-1B Chapter 11, Aircraft Electrical System, section 5 (wiring rating) paragraphs 11-66, 11-67 section 6 (Aircraft Electrical Wire section) paragraphs 11-76, 11-77.

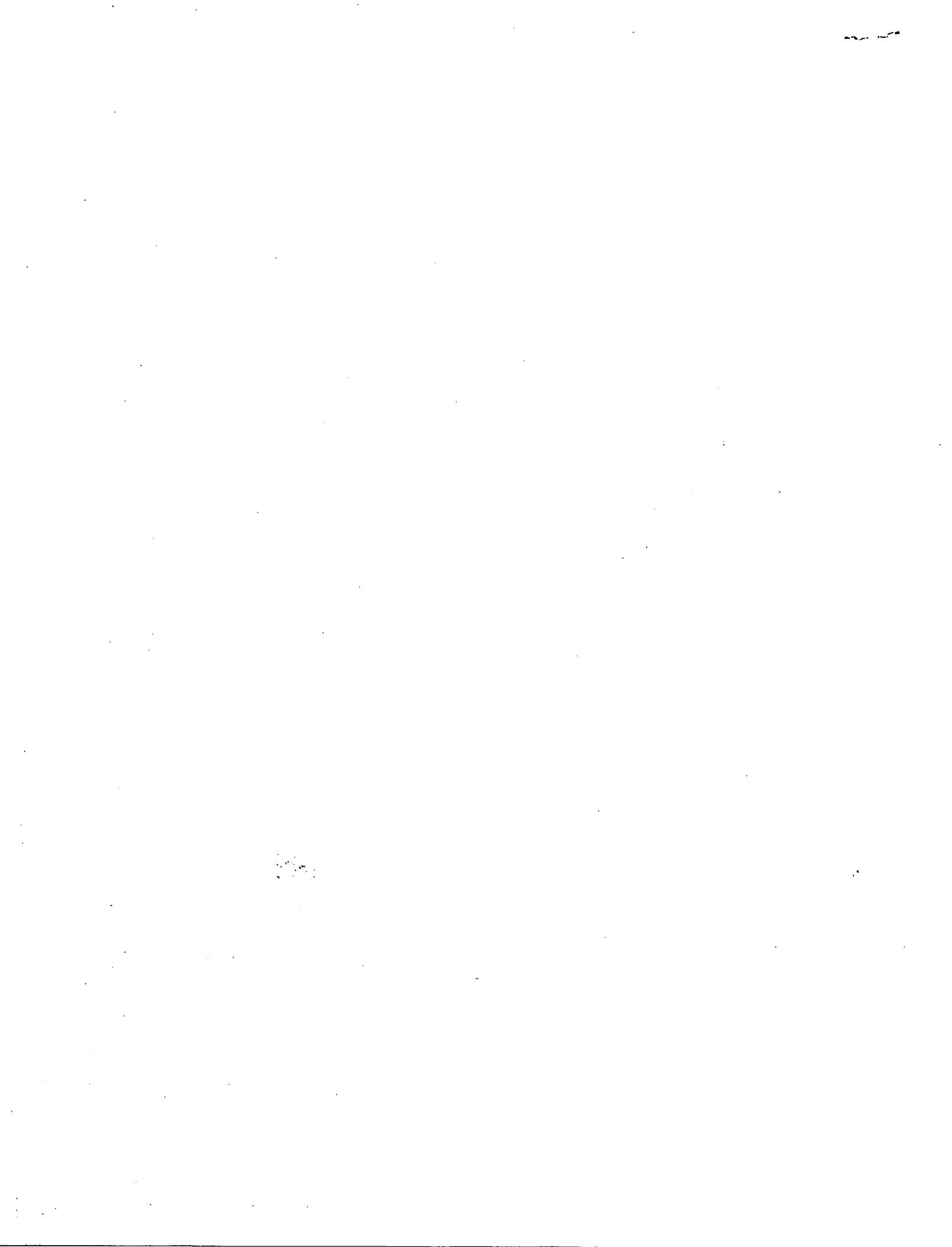
An electrical load does not exceed limitations of AC43-13-1b Chapter 11, paragraphs 424 (Electrical load limits), 425 (generator) and 428 (determination of electrical load).

The Instructions for Continued Airworthiness (ICA) contained in the Flight Standards Handbook Bulletin for Airworthiness (HBAW-8900.1) are not applicable as these components are not field repairable and are "Remove and Replace" items only.

Aircraft weight & balance and equipment list amended as required.

----- Nothing follows -----

Additional Sheets Are Attached



United States of America
Department of Transportation—Federal Aviation Administration

Supplemental Type Certificate

Number SA00295DE

This certificate, issued to
Paravion Technology, Inc.
2001 Airway Avenue
Fort Collins, CO 80524

certifies that the change in the type design for the following product with the limitations and conditions therefor as specified hereon meets the airworthiness requirements of Part 3 of the Civil Air Regulations.

Original Product—Type Certificate Number: A4CE

Make: Cessna Aircraft Company
Model: TU206G, 206H, & T206H

Description of the Type Design Change:

Installation of an external Infrared Imaging System in accordance with Paravion Technology Master Drawing List Report No. DL-C206IR-100, Revision N/C, dated March 29, 1997 or later FAA approved revision.

Limitations and Conditions:

1. This approval should not be extended to aircraft of this model on which other previously approved modifications are incorporated unless it is determined that the interrelationship between this change and any of those other previously approved modifications, including changes in type design, will introduce no adverse effect upon the airworthiness of that aircraft.
2. A copy of this Certificate and Flight Manual Supplement must be maintained as part of the permanent records for the modified aircraft.
3. ~~FAA approved Aircraft Flight Manual Supplement, PR-C206IR-100M, Revision 0, dated June 11, 1997 or later FAA approved revision is required.~~
4. If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission.

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.

Date of application: January 10, 1997

Date reissued:

Date of issuance: June 12, 1997

Date amended: April 8, 2004



By direction of the Administrator

A handwritten signature in black ink that reads "Melissa Sandow".

Melissa Sandow, (Signature) Small Airplane Program Manager
Northwest Mountain Region
Denver Aircraft Certification Office
(Title)

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

This certificate may be transferred in accordance with FAR 21.47.



**PARAVION TECHNOLOGY, INC.
2001 AIRWAY AVENUE
FT. COLLINS, COLORADO 80524**

REPORT NO. PR-C206IR-900M

INSTALLATION INSTRUCTIONS

FOR

INFRARED IMAGING SYSTEM



REVISIONS

<u>REV.</u>	<u>DATE</u>	<u>DESCRIPTION</u>	<u>BY</u>
N/C	11/02/00	Original	MR
A	05/18/01	Added Video Output Note.	MR
B	09/06/02	Added reference to C206IR-101-2 Support Installation, section 2.1.1.	GP
C	10/25/04	Section 2.1.1 added reference to FLIR U8000, U8500 Section 2.1.3 re-worded to clarify doubler installation Added Table 2.2, other minor wording changes to clarify	REB
D	01-07-05	Section 2.2.10 edited to include assembly of Item 33 Doubler and Item 26 beam Assembly.	REB
E	12/06/05	Sect. 2.1.8, page 1 was "... Remove fasteners which conflict with angle installation. Adjust clamps to support tube in center of opening and level tube to cabin floor." Clarified to indicate positioning laterally and longitudinally.	REB

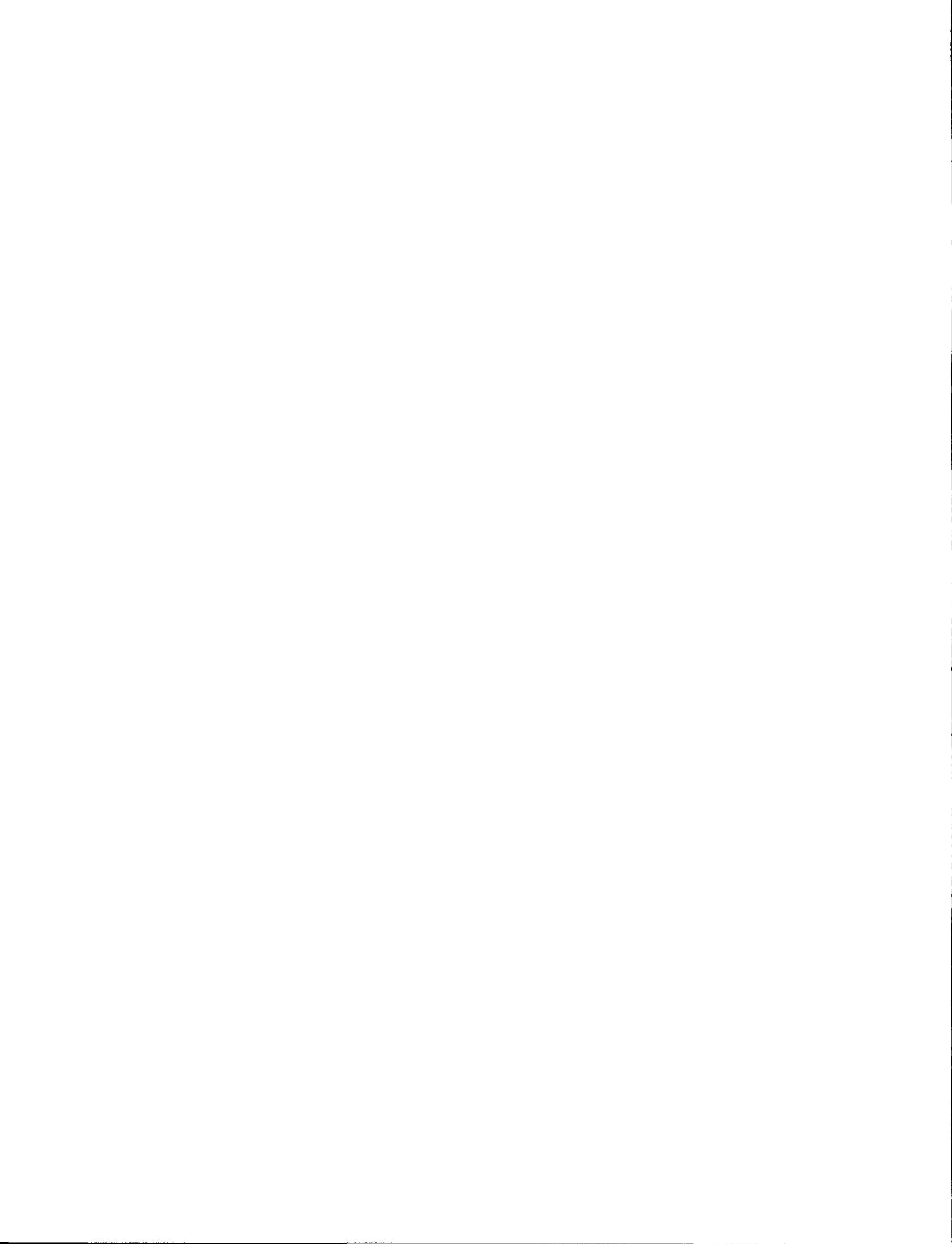


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1.0 INTRODUCTION

This document provides a step-by-step procedure for installation of the C206IR-100 Infrared Camera System Installation in the Cessna 206 Aircraft. The instructions contained herein are intended to supplement the information contained on the installation drawings.

2.0 INSTALLATION PROCEDURES

2.1 Support Installation (Drawing C206IR-101)

- 2.1.1 If your camera system has a dual power/control cable from the gimbal to the electronic control unit (Ref. FLIR MK-I and MK-II) then use C206IR-101-1. If your camera system has a single power/control cable from the gimbal to the electronic control unit (Ref. FLIR MK-III, U7000 and U7500, U8000, U8500) then use C206IR-101-2.
- 2.1.2 Remove baggage floor covering and all necessary side panels.
- 2.1.3 Verify location of doubler between longitudinal stringers in baggage compartment area and trim doubler/shim as necessary. Mark location of doubler on fuselage and remove all conflicting fasteners. Match drill doubler to existing fastener holes. Locate and drill additional fastener holes per drawing. Remove doubler, de-burr holes and install using indicated hardware.
- 2.1.4 Locate and drill indicated hole through both fuselage and doubler as shown, de-burr. Install rivets around hole through fuselage and doubler.
- 2.1.5 Adhere extrusion to circumference of opening.
- 2.1.6 Temporarily clamp angles to support assembly.
- 2.1.7 Position clamped support assembly in aircraft through hole and perpendicular to aircraft centerline.
- 2.1.8 Remove fasteners which conflict with angle installation. Adjust clamps to center the tube in the previously drilled opening and parallel to the baggage compartment floor (laterally). The support tube longitudinal angle should be set by leveling the Electronic Control Unit Mount Bracket to the cabin floor, not to the baggage compartment floor.
CAUTION: The C206IR-2500-1 Spacer (if used) and IR-1030-1 Angle are not symmetrical. Note correct orientation of parts before drilling baggage compartment floor.
- 2.1.9 Mark and match drill floor to support angles, remove clamped assembly.
- 2.1.10 Temporarily install indicated beam Assembly and support angles as shown. Match mark the beam for angles installation. Attach the angles to the beam in accordance to the drawing and temporarily re-install the assembly. Match drill the Beam Assembly to the previously drilled floor. Remove the beam Assembly and install the indicated Doubler (Nut Plate Assembly) using indicated fasteners(**NOTE:** It will be necessary to



trim the width of the doubler to fit inside the beam). Permanently install the beam Assembly using indicated hardware.

- 2.1.11 Match drill support legs to angles and secure using indicated hardware.
- 2.1.12 Reinstall support assembly by securing angles through spacer(optional if needed to adjust height) into the installed fastener assemblies in floor using indicated hardware.
- 2.1.13 Reinstall floor covering and fairing, trimming as necessary.
- 2.1.14 Install placard in a conspicuous location near existing baggage weight limits placard.
- 2.1.15 Optional use of MIL-S-8802F Class B2 sealant and DC4 or equivalent products, as indicated, may be desirable.

2.1 Equipment Cabinet Installation (Drawing C206IR-201)

- 2.2.1 Aircraft built prior to 1997 incorporate a lighter seat rail and require use of the C206IR-201-1 installation. The heavier seat rails in post-1997 aircraft require use of the C206IR-201-2 installation.
- 2.2.2 The equipment cabinet mount plate assembly may be installed to the seat rail pair in place of the copilot seat.

NOTE: See Table 2.2 for available Mount Plate options

TABLE 2.2; EQUIPMENT CABINET MOUNT PLATE INSTALLATIONS

Aircraft Mfr. Date	C206IR-201-1 Equipment Cabinet Installation	C206IR-201-2 Equipment Cabinet Installation
Pre-1997	C182IR-2500-1 Mount Plate Assy. Optional C182IR-2500-3 Assy.	
1997 and Later		C182IR-2500-2 Mount Plate Assy. Optional C182IR-2500-4 Assy.

- 2.2.3 Install Mount Plate Assembly to rail pair in desired location by sliding clamps onto rails. Mark locations for seat pin assemblies and remove to drill indicated holes.
- 2.2.4 Reinstall mount plate assembly to rail, slide clamps tight against rails and tighten screws.
- 2.2.5 Secure FWD/AFT movement by installing seat pin assemblies in drilled holes.
- 2.2.6 Install cabinet by inserting studs on bottom of cabinet into slots in the Mount Plate assembly and slide forward to small end of slot. Secure by inserting bolt through Mount Plate assembly and into cabinet nut plate.
- 2.2.7 Assure all fasteners are securely installed.



NOTE: Weight and balance data must be adjusted in accordance with actual weights and locations of installed equipment.

2.2 Equipment Installation (Drawing C206IR-251)

NOTE: All video outputs to any monitors should come from the VCR if installed.

The monitor and electronics support module (EU) installations are addressed by this drawing.

The equipment cabinet installation is designed to carry up to 25 lb. The equipment mounting bracket for the electronics support module is provided on the C206IR-1010-2 support assembly.

2.3.1 Monitor Installation:

2.3.1.1 The swivel support is designed to carry the Inframetrics monitor. Other monitor installations may require different mounting provisions and separate approval.

2.3.1.2 Remove the top cover of the monitor, then drill and install plate nuts on each side using the indicated rivets.

NOTE: Cover monitor assembly when modifying top cover to keep foreign material out. Reinstall monitor cover.

2.3.1.3 Adhere extrusion to perimeter of monitor glare shield using indicated adhesive.

2.3.1.4 Using indicated hardware, install the swivel support assembly to the monitor.

2.3.1.5 Place the monitor assembly on the top of the equipment cabinet, and secure using indicated hardware.

2.3.2 Electronics Unit Installation:

2.3.2.1 Install IR-2000-1 buttons to EU using indicated hardware.

2.3.2.2 Remove the clip from the rail assembly of EU mount.

2.3.2.3 Move the sliding shafts away from the keyholes in the rail assembly.

2.3.2.4 Fit the buttons on the EU into the keyways, and slide them into the keyway slots.



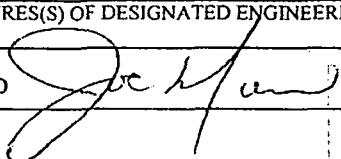
2.3.2.5 Move the sliding shafts against the EU and tighten the lock knobs. Replace the clip.

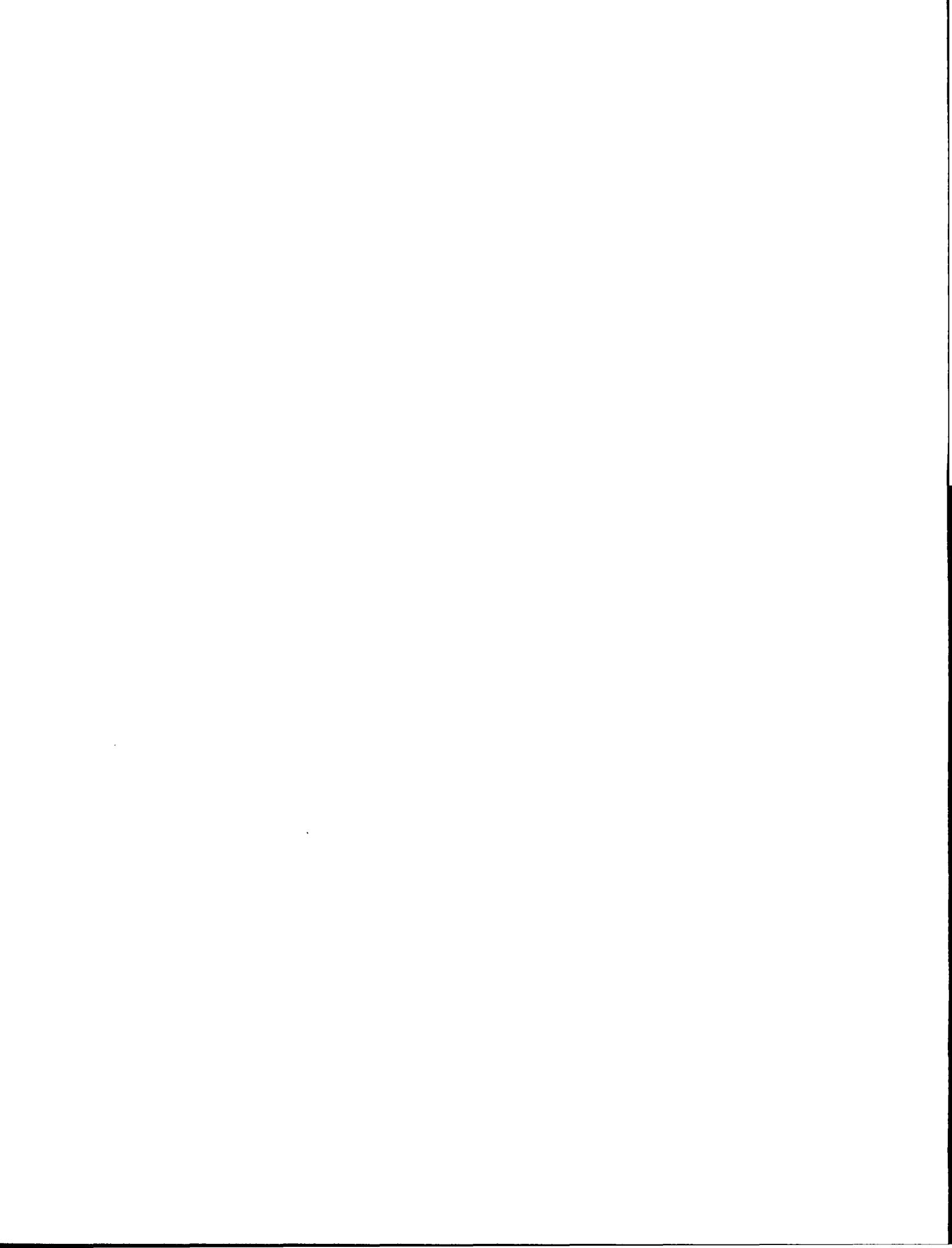
NOTE: Weight and balance calculations must include installed equipment.

2.4 Power Supply Wiring (Drawing C206IR-301)

- 2.4.1 Refer to drawing and camera system specification data for cable identification and connection.
- 2.4.2 Remove panels as necessary.
- 2.4.3 Install indicated circuit breaker in available aircraft breaker panel position. Provide electrical power through avionics buss.
- 2.4.4 Locate unused rocker type switch in lighting panel for use as infrared on/off switch.
- 2.4.5 Route power cable to electronics unit from infrared on/off switch. General cable routing should follow existing electrical wiring.
- 2.4.6 When system is installed for use, loose cables should be routed under seats and otherwise secured.
- 2.4.7 When system is disabled or removed, loose cables and controls must be stowed or removed.

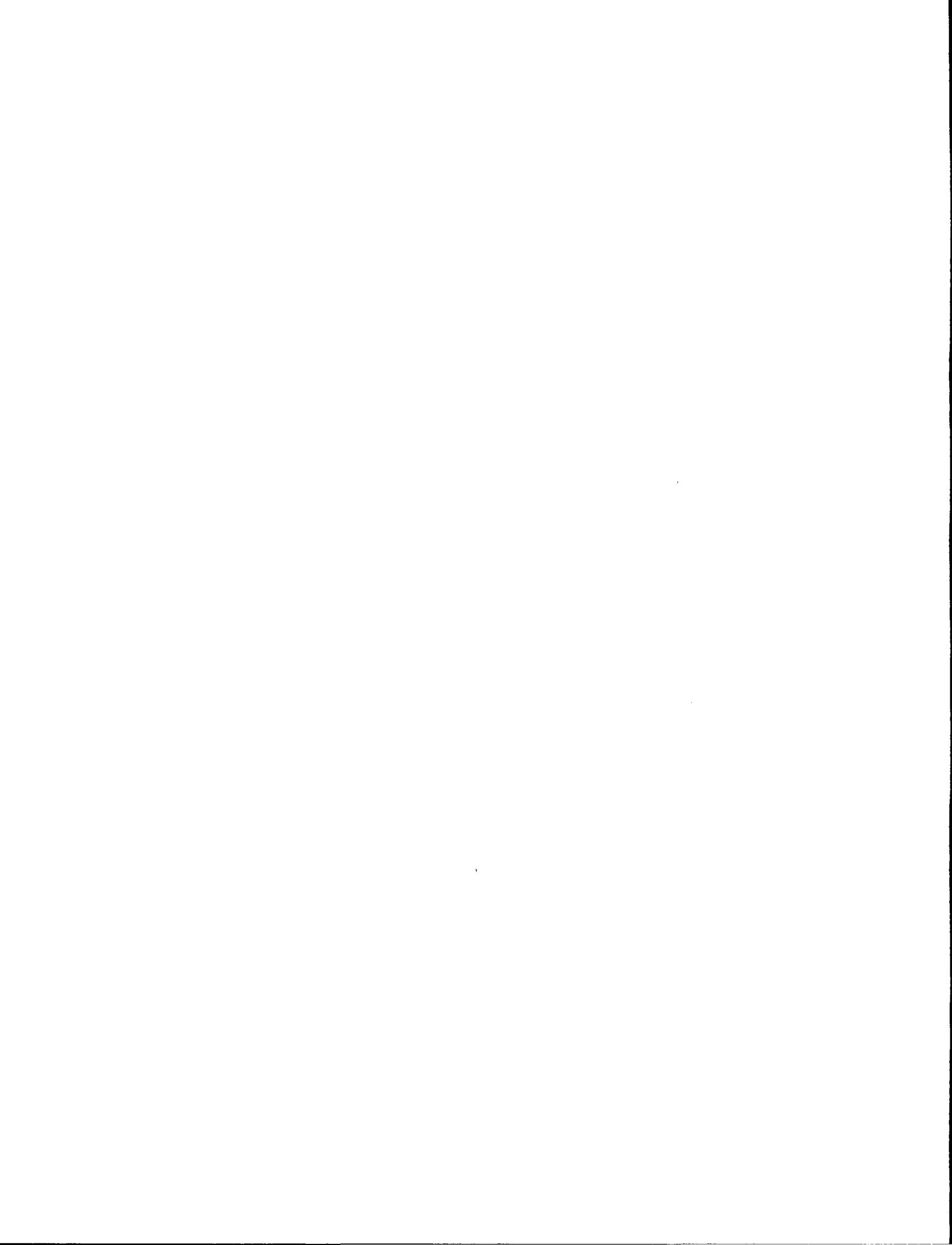


U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS				1. DATE 12/28/2012
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION				
2. MAKE Cessna Aircraft Company	3. MODEL NO. 206	4. TYPE (Airplane, Radio, Helicopter, etc.) Airplane	5. NAME OF APPLICANT Paravion Technology Inc.	
LIST OF DATA				
6. IDENTIFICATION Paravion Technology Drawing IR-605 Rev A dated 11/13/2012 Paravion Technology Document ER-C206ELP-2 (MX-10 Installation) Rev 4 dated 12/21/2012	7. TITLE Support Plate Structural Analysis for Wescam MX-10 Installation Using 206IR-101 Support Installation & IR-605 Support Plate			
<p>Notes:</p> <ol style="list-style-type: none"> Only structural aspects of the above data are approved herein. This approval is for engineering design data only and is not an installation approval. It indicates the data listed above demonstrates compliance with the regulations specified by paragraph and subparagraph listed below as 'APPLICABLE REQUIREMENTS'. (Compliance to additional regulations not listed here may be required). This form does not constitute FAA approval of all engineering data necessary for substantiation of compliance to necessary requirements for the entire alteration/repair. This approval is valid for Cessna Aircraft Company Model 206 S/N T20608983 				
8. PURPOSE OF DATA Submittal of data in support of FAA Major Alteration				
9. APPLICABLE REQUIREMENTS (List specific sections) 14 CFR 23.301, 23.303, 23.305(a), 23.307(a), 23.337(a), 23.601(a), 23.603, 23.605(a), 23.607(a), 23.609(a)(2), 23.611, 23.613, & 23.625(a).				
10. CERTIFICATION -Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183 of the Federal Aviation Regulations, data listed above and on attached sheets numbered <u>2</u> have been examined in accordance with established procedures and found to comply with applicable requirements of the Federal Aviation Regulations.				
<input type="checkbox"/> Therefore Recommend approval of these data <input checked="" type="checkbox"/> Approve these data				
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S) 		12. DESIGNATION NUMBERS(S) DERT-605388-NM	13. CLASSIFICATION(S) Structures	
FAA FORM 8110-3 (03/10) SUPERSEDES PREVIOUS EDITION				



Applicable Requirement Amendment Levels:

FAR	Title	Amdt.
23.301	Loads	23-42
23.303	Factor of safety	-
23.305(a)	Strength and deformation	-
23.307(a)	Proof of structure	-
23.337(a)	Limit maneuvering load factor	-
23.601(a)	Design	-
23.603	Materials	-
23.605(a)	Fabrication methods	-
23.607(a)	Fasteners	-
23.609(a)(2)	Protection of structure	-
23.611	Inspection provisions	23-7
23.613	Material strength properties and design values	-
23.625(a)	Fitting factors	-



FLAMMABILITY REPORT, TEST, AND TEST RESULTS

**PARAVION TECHNOLOGY, INC.
2001 AIRWAY AVENUE
FORT COLLINS, CO 80524**



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13. APPENDIX A: TEST RESULTS REPORT FROM MR. LAZAROFF	A-1
14. APPENDIX B: FAA FORM 8110-3	B-1



flammability tests described in this document. The Vertical Flammability Test Cabinet meets the requirements of 14 CFR Part 25, Appendix F, Part I, Section (b)(3) and DOT/FAA/AR-00/12, Aircraft Materials Fire Test Handbook.

Equipment Nomenclature	Model Number	Manufacturer
Conditioning Chamber	Stabil-Therm Laboratory Oven	Blue M Electric Co. Blue Island, IL
Vertical Flammability Test Cabinet	7633A	United States Testing Co., Inc. Hoboken, NJ

5. TEST ARTICLE CONDITIONING

The test articles will be conditioned to $70^{\circ} \pm 5^{\circ}\text{F}$ and at $50\% \pm 5\%$ relative humidity until moisture equilibrium is reached or for 24 hours. Each specimen must remain in the conditioning environment until it is subjected to the flame.

6. CONFORMITY INSPECTIONS

Company conformity of the test articles will be conducted by Paravion Technology and documented on FAA Form 8130-9, Statement of Conformity. The test set-up will be verified by the witnessing Flammability DER to be in accordance with this test plan. The witnessing DER will coordinate with Paravion Technology, Inc, if design data changes are necessary, prior to DER approval.

7. VERTICAL TEST SET UP CONFIGURATION

The vertical test will be configured and conformed as follows, in accordance with 14 CFR Part 25, Appendix F, Part I, Section (b)(4):

- A Bunsen burner with a nominal 3/8 inch I.D. tube will be used for the test.
- Prior to testing, ignite the burner and set the flame height to 1½ inches.
- Using a calibrated thermocouple pyrometer, verify that the minimum flame temperature in the center of the flame is 1550° F . Record the flame temperature on the Test Data Sheet in Appendix B, and extinguish the flame.
- Set the automatic timer on the Flame Control Module to 12.0 seconds.
- Verify that the conditioning chamber has maintained the test articles at $70^{\circ} \pm 5^{\circ}\text{F}$ and $50\% \pm 5\%$ relative humidity for a minimum of 24 hours.
- Remove one test article from the conditioning chamber.



- Verify the test article has been positioned in the chamber in accordance with Section 7.0.
- Activate the automatic flame timer switch.
- Verify that the burner ignites and the flame is applied to the center of the edge of the specimen.
- Verify that the automatic flame timer extinguishes the flame after 12 seconds.
- Observe the behavior of the specimen after the burner flame is extinguished. Continue timing as long as the specimen continues to flame. Note any drippings from the specimen and the flame time of the drippings.
- Record the flame time, burn length (to the nearest 0.1 inch) and flame time of drippings on the test data sheet.

9. PASS/FAIL CRITERIA

The material is considered to pass this test if all of the following criteria are met:

- The average flame time of the specimen after removal of the flame source may not exceed 15 seconds.
- The average burn length may not exceed 8 inches.
- Drippings from the test specimen may not continue to flame for more than an average of 5 seconds after falling.

10. TEST WITNESSING AND DATA APPROVAL

The selected Flammability DER will witness the tests and approve the test results. A copy of the approved test report and 8110-3 will be forwarded to Paravion Technology, Inc.

11. DESIGNATED PERSONNEL

The following is a list of designated personnel to be involved in this project:

Title	Name
DER Flammability	Bob Lazaroff DERT-660022-NM

12. REFERENCES

The following documents form a part of this document to the extent specified herein:



13. APPENDIX A: TEST RESULTS REPORT FROM MR. LAZAROFF



Cessna 206
Console

FLAMMABILITY TEST REPORT
DATA SHEET

C206-CF Console

Test Article ID/PN	Material	Representative of: (list part number and description)	Regulatory Requirement and Amendment Level	Test Criteria per Part 25 Appendix F, Part I (mark "X" in appropriate block)			
				(a)(1)(i) 60 second Vertical	(a)(1)(ii) 12 second Vertical	(a)(1)(iv) 15 second Horizontal	(a)(1)(v) 15 second Horizontal 4.0 in/min
1, 2, 3	1x1 Plain Weave Carbon 2x2 Twill 3K Carbon LAM-135-FR Resin LAM-229 Hardener	Console	23.853(a) Amdt 23-62 *		X		

Test Sample	Test Results per Part 25 Appendix F, Part I (enter results in appropriate block)							
	(a)(1)(i) 60 second Vertical			(a)(1)(ii) 12 second Vertical			(a)(1)(iv) 15 second Horizontal	(a)(1)(v) 15 second Horizontal
	Burn Length (< 6 in)	After-Flame (< 15 sec)	Drip Flame (< 3 sec)	Burn Length (< 8 in)	After-Flame (< 15 sec)	Drip Flame (< 5 sec)	Burn Rate (< 2.5 in/min)	Burn Rate (< 4.0 in/min)
1				1.75 in	10.8 sec	No drips		
2				1.75 in	11.1 sec	No drips		
3				1.0 in	2.9 sec	No drips		
Average				1.50 in	8.3 sec	No drips		

* Tested to Part 25 requirements

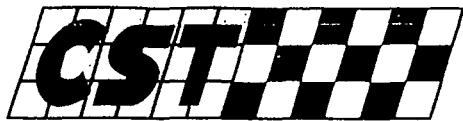
Results: PASS FAIL

DER: Robert Claryeff DEET-660022-NM



U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS				1. DATE 04/10/2013
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION				
2. MAKE Cessna	3. MODEL NO. T206	4. TYPE (Aircraft, Engine, Propeller, etc.) Aircraft	5. NAME OF APPLICANT Paravion Technology, Inc.	
LIST OF DATA				
6. IDENTIFICATION	7. TITLE			
Report No. C206-CF Console Revision 1, 04/08/2013	Flammability Report, Test and Test Results			
	<p>Notes:</p> <p>1. All engineering aspects of the above listed data are approved herein. This approval is for engineering data only. It indicates the data listed above demonstrates compliance only with the regulation specified by paragraphs and subparagraph listed below as "Applicable Requirements".</p> <p>2. This approval is for flammability aspects of the proposed installation only. Additional approvals may be required for the substantiation of compliance to necessary requirements for the entire type design change.</p> <p>3. Delegation to approve test plan and witness tests by Satish Lall, Denver ACO, per Special Authorization, 4/1/13.</p> <p>4. This approval is valid for the installation of the Hand Control Connector Housing on Cessna Model T206 S/N T20608983 only.</p>			
8. PURPOSE OF DATA	In support of a major alteration to Cessna Model 206 S/N T20608983			
9. APPLICABLE REQUIREMENTS (List specific sections)	14 CFR 23.853(a), Amendment 23-62			
10. CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183, data listed above and on attached sheets numbered <u>N/A</u> have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards listed.				
<input type="checkbox"/> Recommend approval of these data I (We) Therefore <input checked="" type="checkbox"/> Approve these data				
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)	12. DESIGNATION NUMBERS(S)	13. CLASSIFICATION(S)		
Robert C. Lazaroff 	DERT-660022-NM	Structures		





The Composites Store, Inc.

COPY

PRODUCT CONFORMANCE CERTIFICATION

Customer No.: R17364

Ship Date: 03/11/2013

Sold To: PARAVION TECHNOLOGY, INC. Ship To: SAME

Purchase Order No. 43778

Invoice No.: 130964

Item No.	Description	Quantity
CF141	3.5 oz. Carbon Fabric, 42" wide, Plain Weave	9 ft.
FDI Style Number: 824		
Lot Number: 22032	Yarn Type: T300 1K_309_NT	
Roll Number: 6A	Date of Mfg: 08/2012	
Weave: PW,	Count: 24.1 x23.2 inches; Width: 42 ¾ "; Weight: 126.0 g/m ²	
Thickness: 0.008 inches		

We hereby certify that the material listed conforms to applicable commercial specifications, or government specifications as shown below.

SPECIFICATION: Commercial Grade

Gail Gewain, President

Website: www.cstsales.com

P.O. Box 622, Tehachapi, California 93581-0622 • Phone: 661-822-4162 • Fax 661-822-4121



U.S. COMPOSITES, INC.
561/588-1001
6670 WHITE DRIVE
WEST PALM BEACH FL 33407

* * * Invoice 256020 * * *

Bill To :
Paravion Technology, Inc.
Valerie McAlpine
2001 Airway Ave.
FORT COLLINS 80524

Ship To : Customer No. 270051

DATE	ORDER NO.	SLS.NO.	ORDER DATE	PURCHASE ORDER	SPECIAL INSTRUCTIONS	
03/28/13			03/28/13			
Quantity U/M	Description Stock Number			Code	Price	Extension
1.00 ea	CERT FEE FOR PREVIOUS SHIPMENT INVOICE # 255095			0004	\$10.00	\$10.00
	LOT # 1233808 ROLL # 0003802464					

Payment/Terms : SUBTOTAL : \$10.00
prepaid credit card TAX :
FREIGHT :
INVOICE TOTAL DUE ===> \$10.00



From the Quality Control Laboratories of:



U S COMPOSITES INC
6670 WHITE DRIVE
WEST PALM BEACH FL 33407

Date: 01/29/13 9:35:25
Page: 2
Style: 94933
Width: 50.0
Code/Part#
Contract: KB00219873 10

Attention:

Cust P.O.: 403389
Weave: 2X2 TWILL

Lot Number: 01233808/00010

Yarn Type Warp: TR30S 3K

Fill: TR30S 3K

Lot #	Cut	Roll #	Shipment Defect Summary		
			Yards	Distortion	Yard Length Defect
1233808	009	0003802303	100	.1	6 DISTORTION
1233808	011	0003802308	100	.0	No Defects Recorded
1233808	013	0003802463	100	.1	No Defects Recorded
1233808	012	0003802464	100	.0	No Defects Recorded
1233808	017	0003803687	100	.2	No Defects Recorded
1233808	016	0003803688	100	.0	No Defects Recorded





GRAFIL INC.

Certificate of Conformity

6900 88th Street
Sacramento, CA 95828 USA

Order No. 30586

Customer No. BGF01

Phone No. 916/386-1733 or 800/365-5633
Fax No. 916/379-2183

Deliver To: BGF Industries Inc.
Cheraw Speciality Weaving
90 Huger Street
90 Huger Street
Cheraw, SC 29520
USA

Certificate of Conformity: 23444

GI Reference: 30586

Certificate Date: 01/16/2013

Fiber Type: TR30S 3K 1.2%S
PYROFIL

Size: 1.2%S

Quantity In lbs: 3,491.84

Customer Purchase Order		Item #	Specification			Salesperson	Customer Part #	
03006103-20		TR-30S-3LSL	CF-202 Ver.1			Wayne Schaefer		
Batch No.	Date of Manufacture	Quantity (lbs)	Strength (ksi)	Modulus (mci)	Yield (yd/lb)	Fiber Density (lb/in ³)	Size Content (% by Mass)	Elongation (%)
1233303A	03/2012	357.12	604.0	33.8	2400.0	0.06429	1.2	1.80
1263304A	05/2012	3,134.72	603.0	33.4	2400.0	0.06419	1.2	1.80
	Cert total:	3,491.84						

Shelf Life for Pyrofil: 3 years from Date of Manufacture

Certified that the supplies/services detailed herein have been inspected and tested in accordance with the conditions and requirements of the contract or conform in all respects to the specification(s), drawings relevant thereto.

Signed: _____

For and on behalf of
Grafil Inc.



Monday, March 11, 2013

PRO-SET

To Whom it May Concern:

This is to certify that the PRO-SET® product(s) that you recently purchased were manufactured in accordance with our standard quality control procedures. In addition, a representative sample from each batch was tested for conformity to our internal specifications and a portion of that sample will be retained for 18 months from the production date.

RESULTS OF ANALYSIS

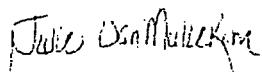
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LAM-135-FR	1363018A	01-18-2013	LAM-224
<u>DSC 822e</u>		<u>Limits</u>	
TPeak1=85.69(°C)		85.38-86.73(°C)	
TPeak2=0(°C)		-°C)	
dH=489.44(J/g)		434.82-505.34(J/g)	
tg=90.43(°C)		89.77-93.4(°C)	
<u>Viscosity</u>		<u>Limits</u>	
3972cPs		3662-4027cPs	

STATEMENT OF SHELF LIFE

Minimum 3 years from date of manufacture when stored in original sealed container.

Sincerely,

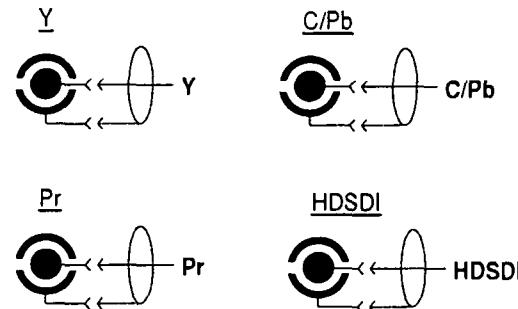
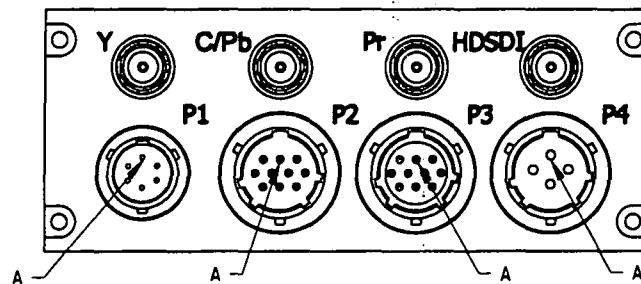
PRO-SET, INC.



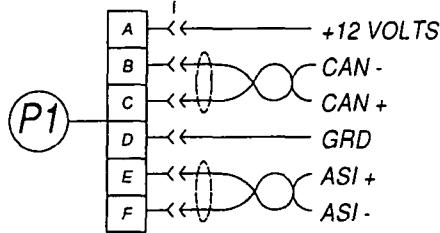
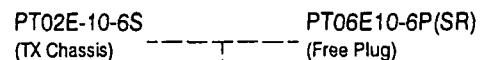


NOTES:
VOLTAGE RANGE: 11 - 30 VOLTS
POWER CONSUMPTION: 72 VA
TROLL INTERFACE: CONNECTOR P2 PINS H, J & K
FOR REFERENCE USE ONLY

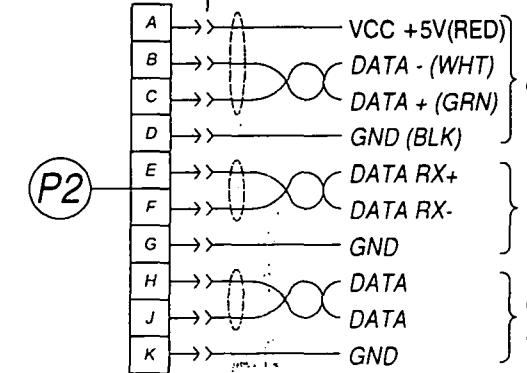
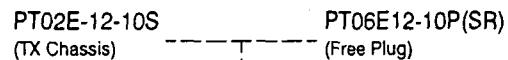
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REV	DESCRIPTION	DATE	APPROVED
A	INITIAL RELEASE	3/16/2011	O REYES



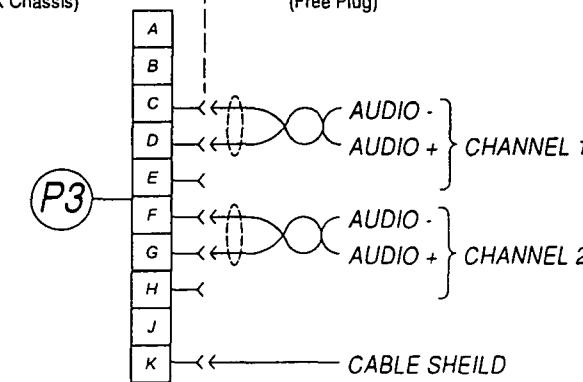
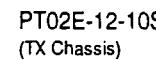
HAND HELD CONTROLLER



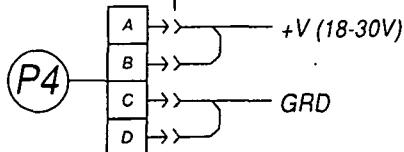
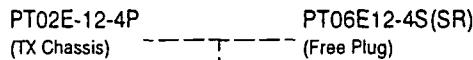
DATA INTERFACE



AUDIO INTERFACE



POWER INTERCONNECT



TOLERANCE INFO:
ALL DIMENSIONS ARE IN INCHES.
DECIMAL .XX ±.01, .XXX ±.0025, ANGULAR ± .3°
HOLE SIZE +.003, -.002, CONCENTRICITY ±.001.

CONFIDENTIAL INFO:
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INC. AND MAY NOT BE REPRODUCED OR DISTRIBUTED
WITHOUT THE WRITTEN CONSENT OF JANTEC INC.

PART NO:

REV: DESIGNER:

J PO

ENGINEER:
ORE

SHEET: DATE:
1 OF 1 3/16/

AVIATION

AVIATION INTERVIEW

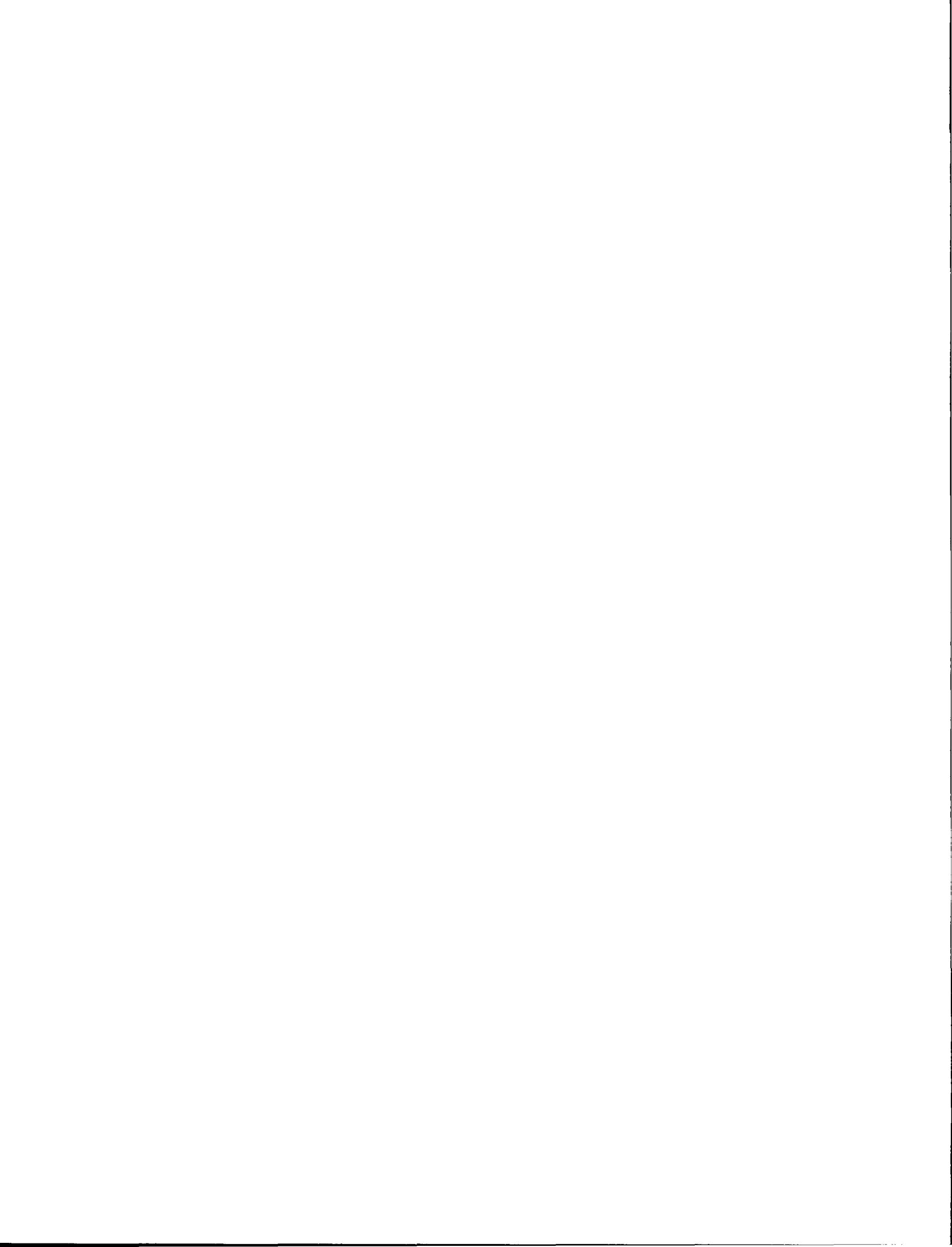
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JANTEC

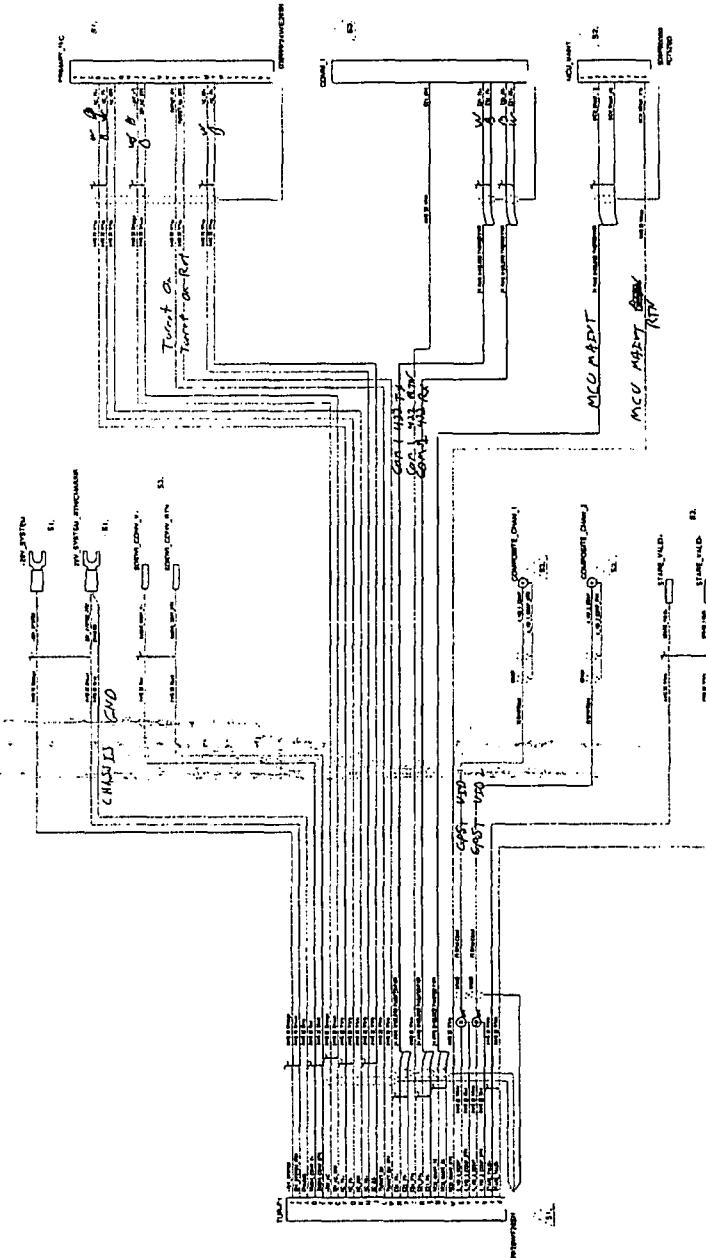
SAN FRANCISCO
CORPORATION

AVIATION Tx WIRING INTERFACE

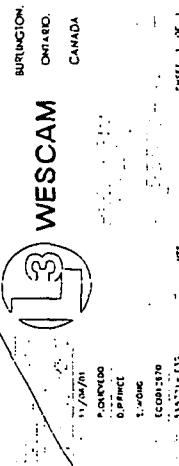
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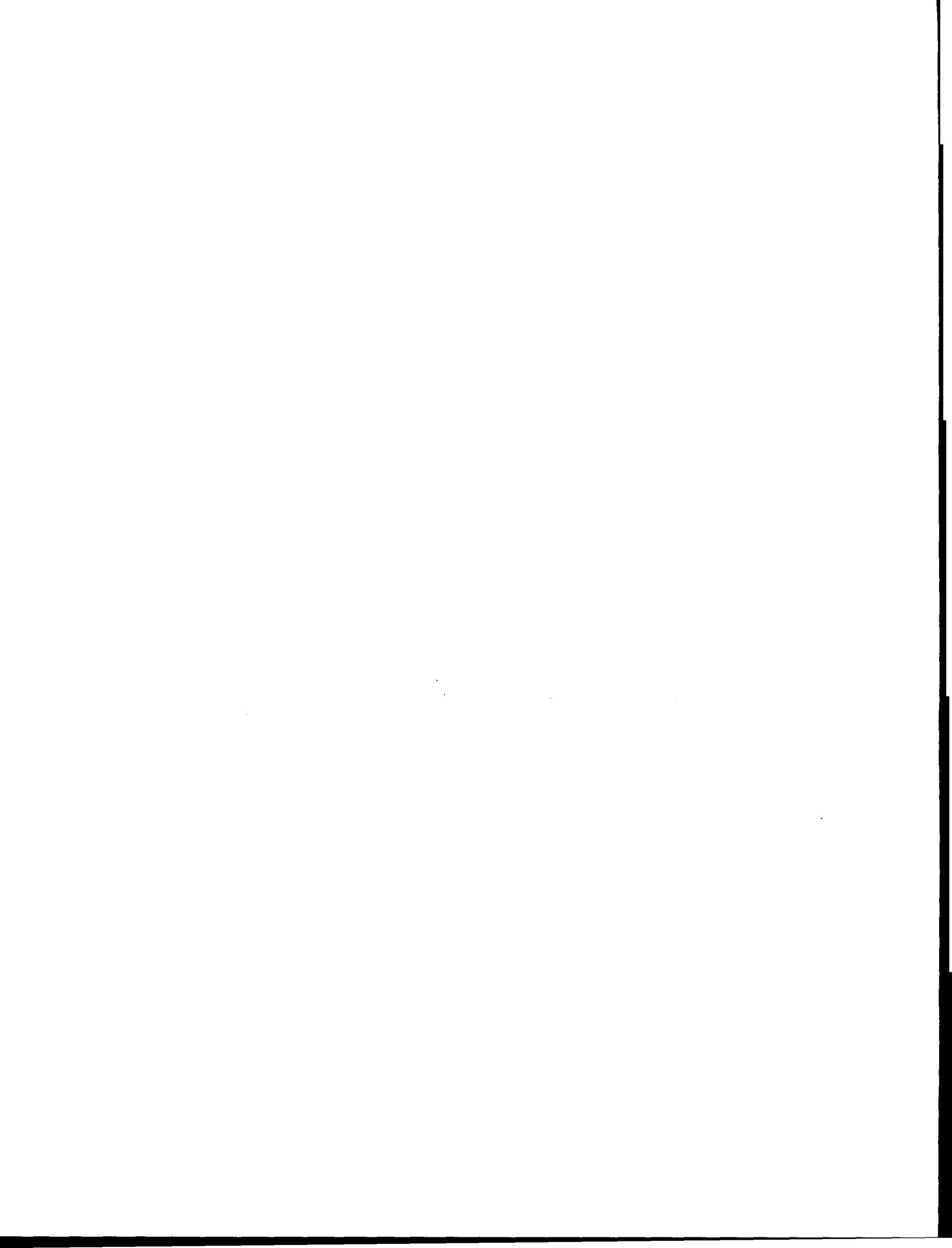
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 DRAWN: 11/14/01
 CHECKED: 11/14/01
 APPROVED: 11/14/01
 BY: D. SPARRETT



- NOTES:**
- SPECIFIED CONNECTOR IS REQUIRED.
 - RECOMMENDED CONNECTOR WOULD BE USEFUL FOR WESCAM TROUBLE SHOOTING. HOWEVER IT IS NOT REQUIRED.
 - OPTIONAL REQUIREMENT DEPENDANT ON INDIVIDUAL SYSTEM CONFIGURATION AND OPTIONS. SPECIFIED CONNECTOR IS REQUIRED IF CONSTRUCTION IS DESIRED.
 - ENSURE GOOD ELECTRICAL CONDUCTIVITY BETWEEN SHELLS AND BUCKSHELL.
 - MINIMUM WIRE GAUGE IS RECOMMENDED.
 - REFERENCE DOCUMENT: 4129 CIO FOR K110 DIGITAL TURRET.
 - REFERENCE DOCUMENT: 4129 CIO FOR K110 DIGITAL TURRET.



Sheet 1 of 1



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AUTHORIZE BY WESCAM (IN WRITING).

SAC24 53574-01 1 -

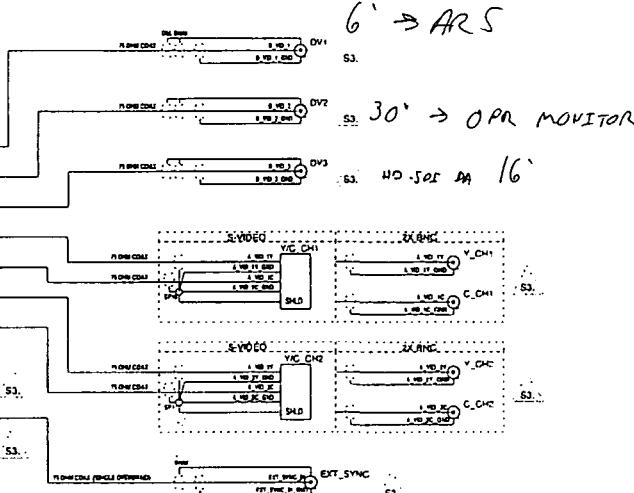
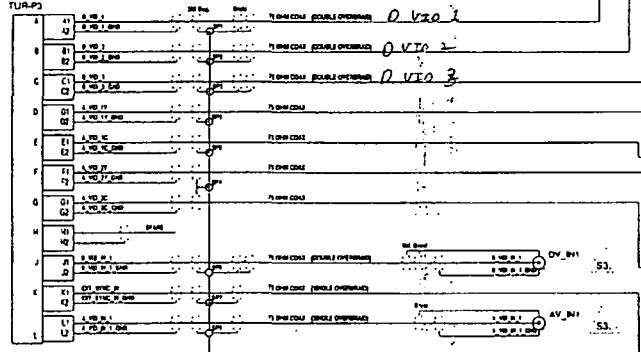
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DO NOT SCALE DRAWING

ZONE : REV.

REVISIONS
DESCRIPTION

DATE APP'D

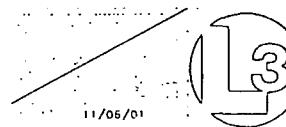
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D3899M25WF1168
M39029/77-428 S1.

NOTES:

- S1. SPECIFIED CONNECTOR IS REQUIRED.
- S2. RECOMMENDED; CONNECTOR WOULD BE USEFUL FOR WESCAM TROUBLE SHOOTING, HOWEVER IT IS NOT REQUIRED.
- S3. OPTIONAL; REQUIREMENT DEPENDANT ON INDIVIDUAL SYSTEM CONFIGURATION AND OPTIONS. SPECIFIED CONNECTOR IS REQUIRED IF CONFIGURATION IS DESIRED.
- ENSURE GOOD ELECTRICAL CONDUCTIVITY BETWEEN SHIELDS AND BACKSHELL.
- MINIMUM WIRE GAUGE IS IDENTIFIED.
- REFERENCE DOCUMENT: 84270 ICD FOR MX-10 DIGITAL TURRET.
- REFERENCE ONLY, ICD TAKES PRECEDENCE.



WESCAM

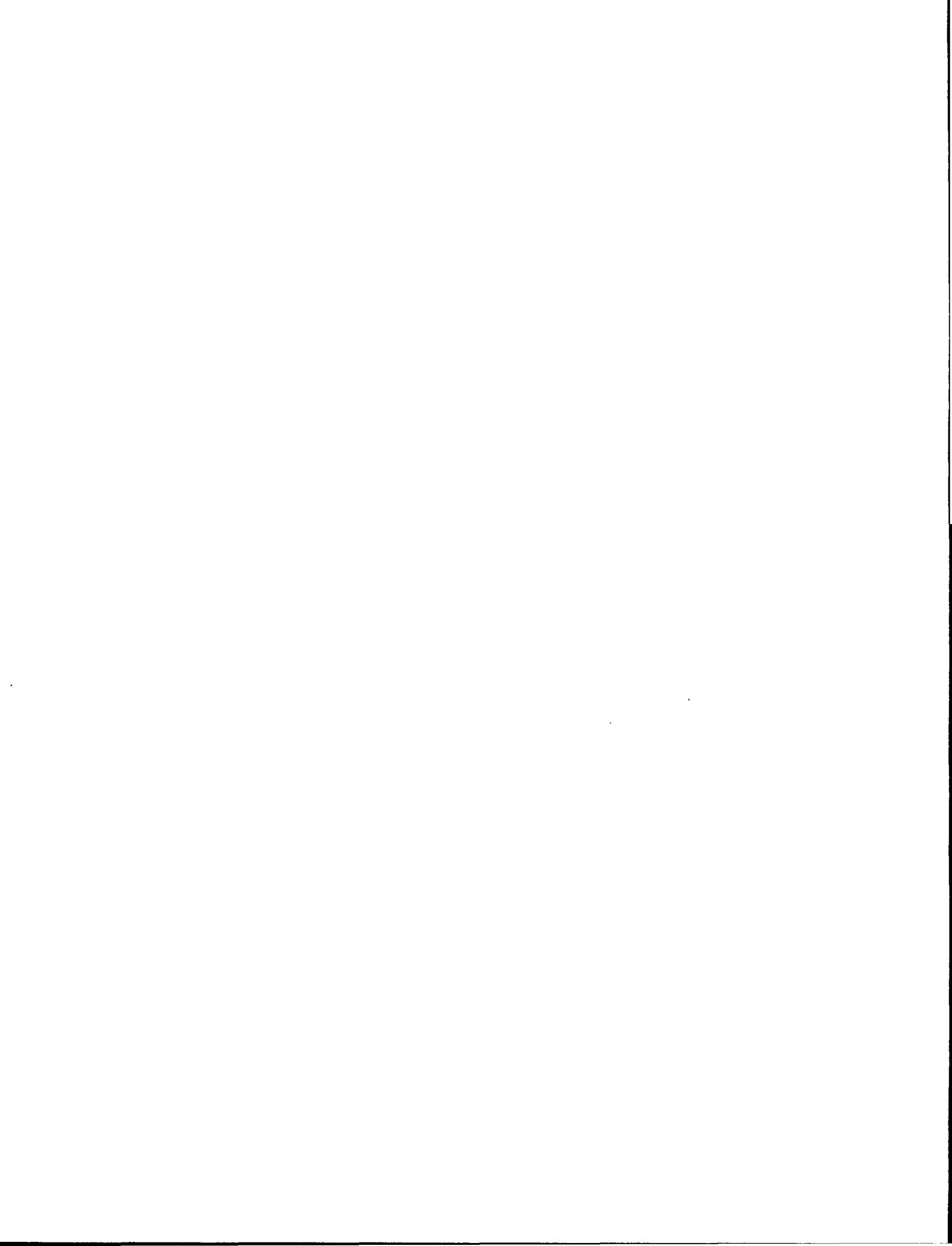
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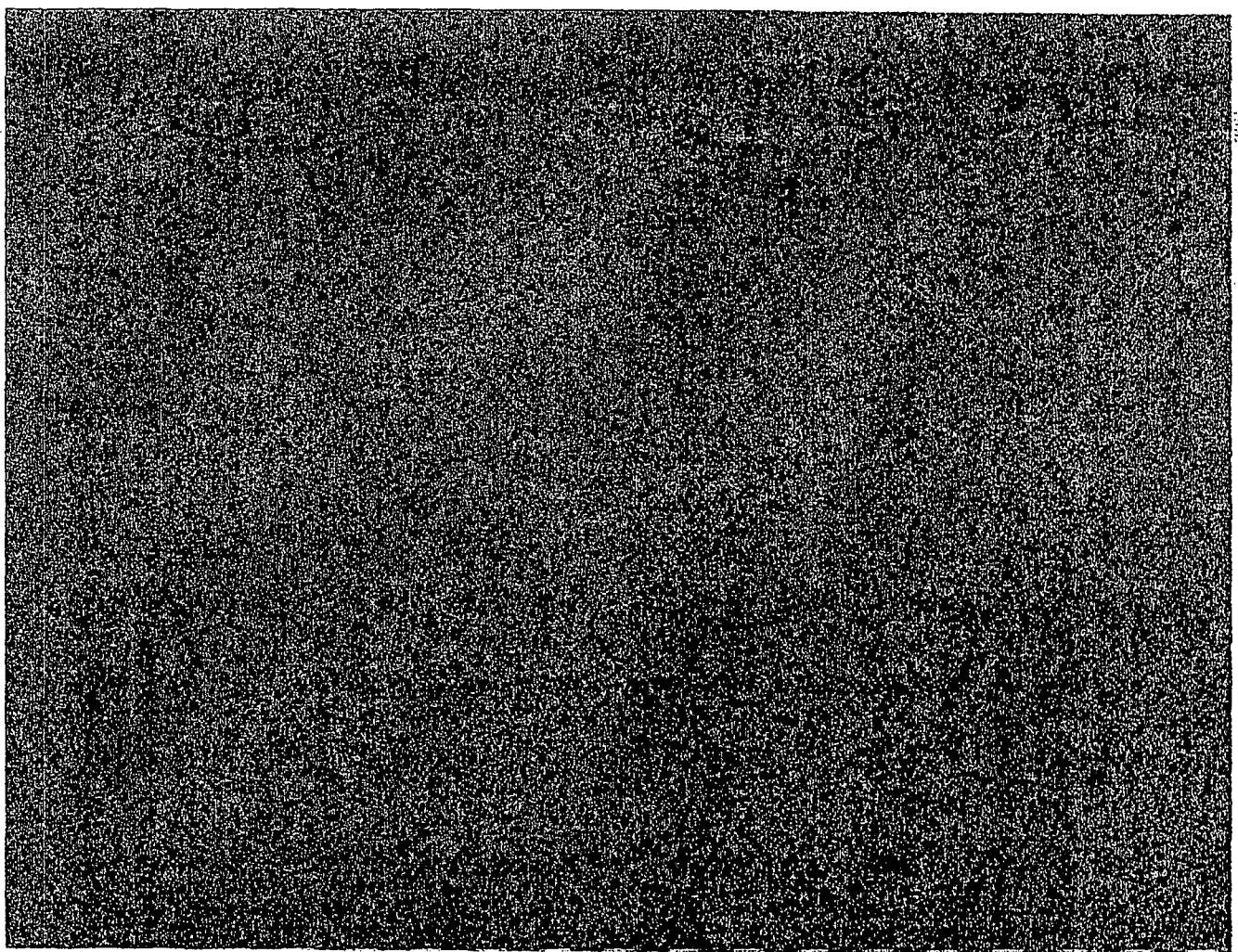
11/06/01
P.QUEVEDO
D.PRINCE
S.WONG
EC0012670

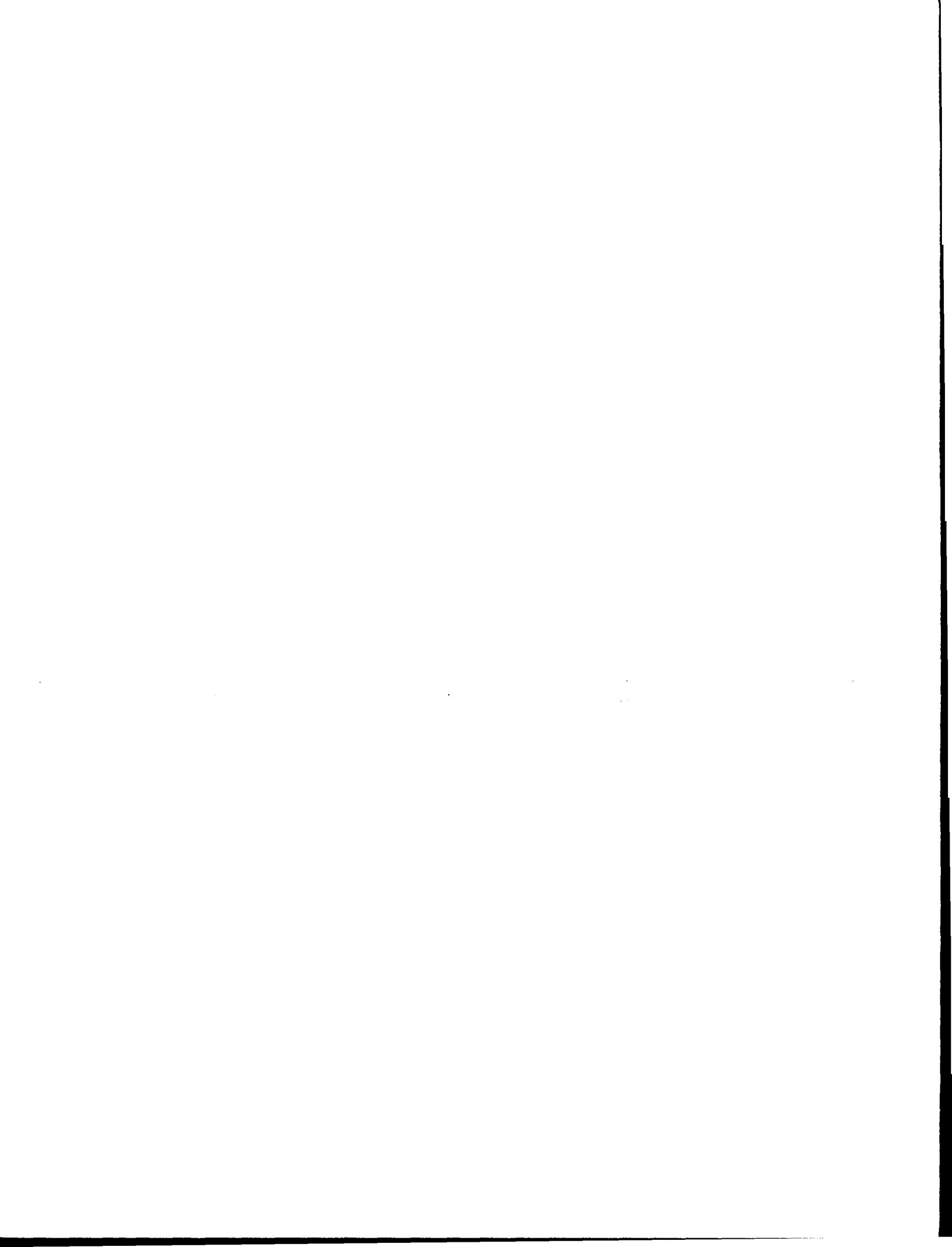
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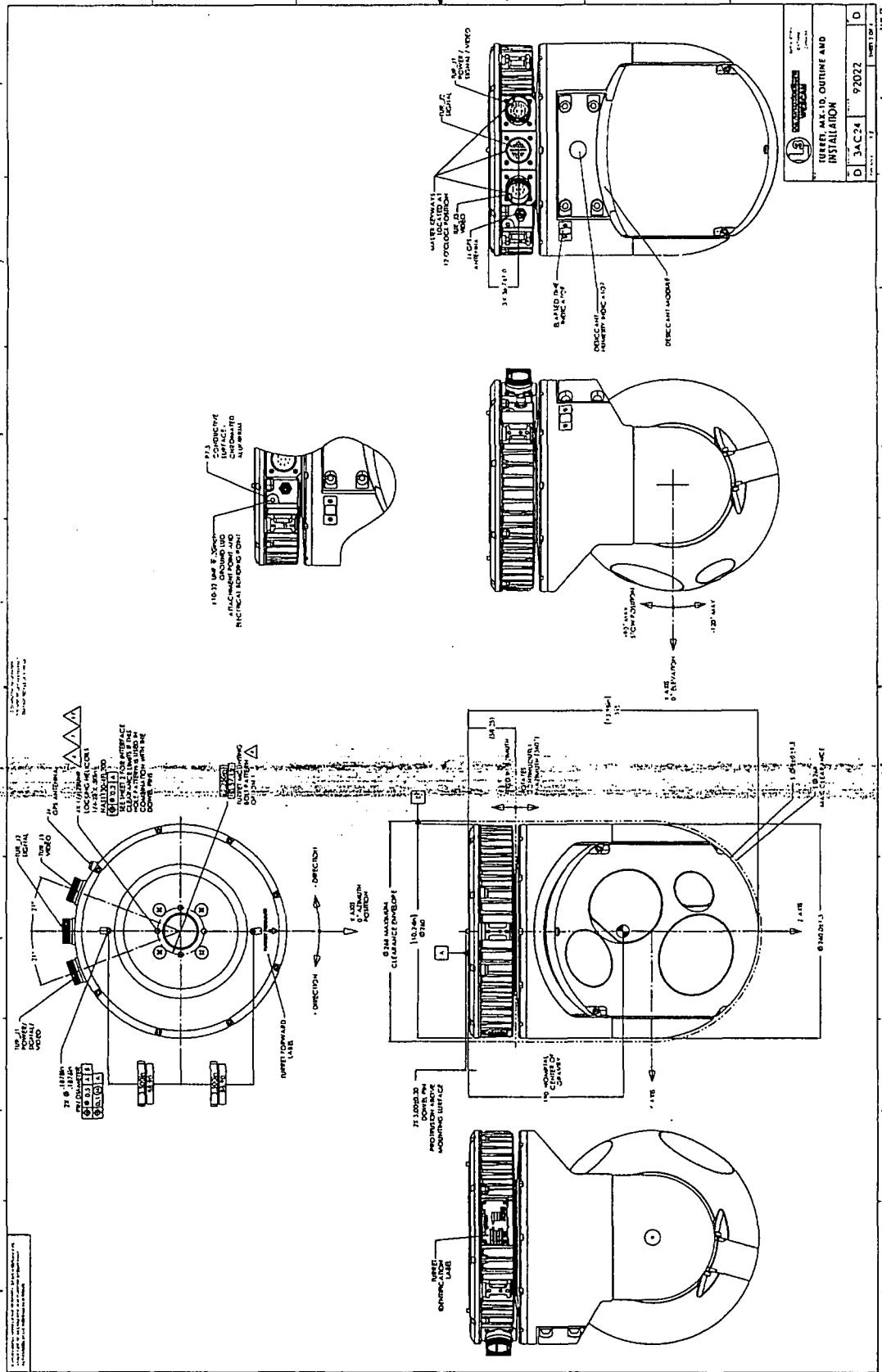
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SHEET 1 OF 1

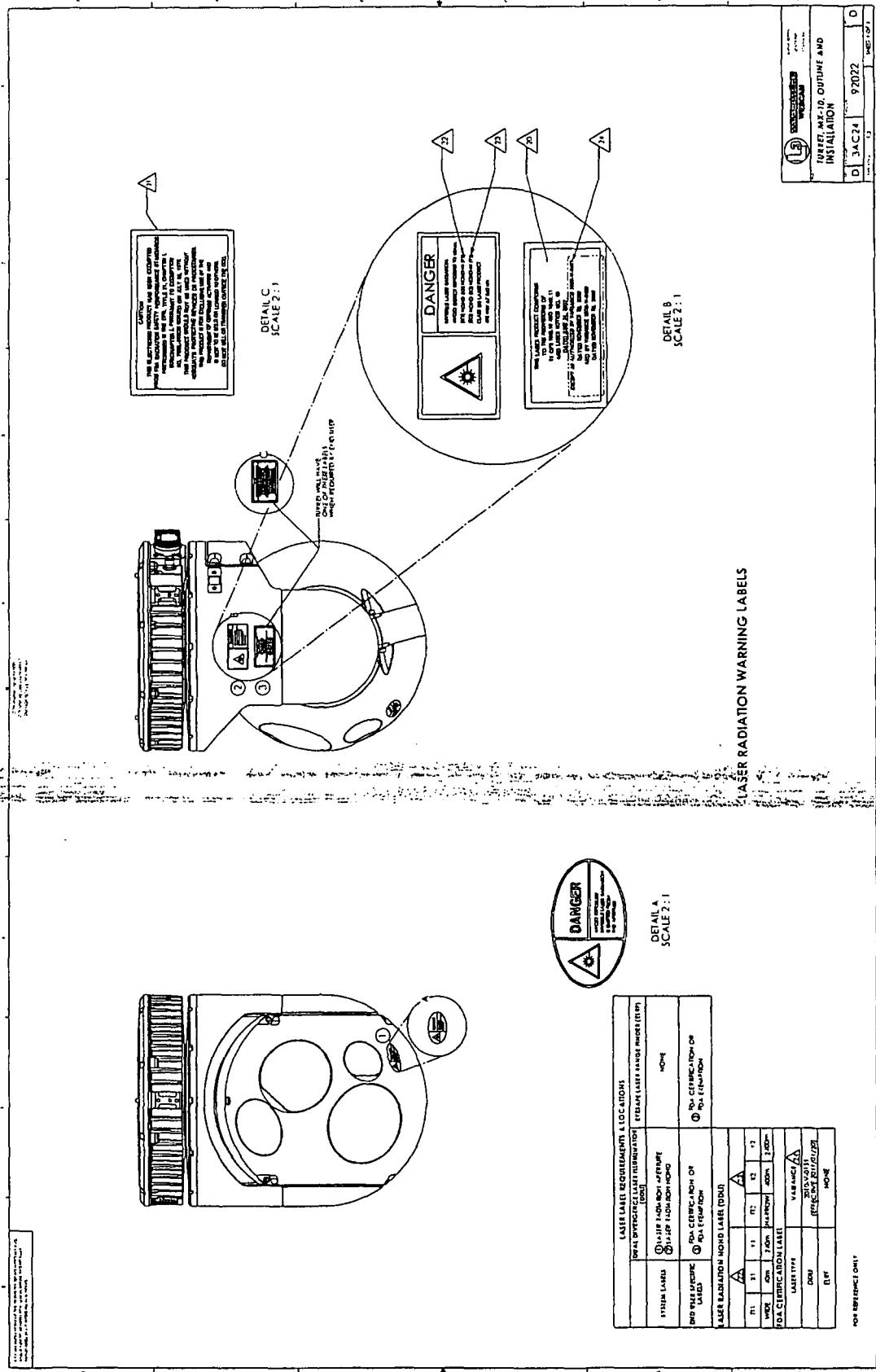




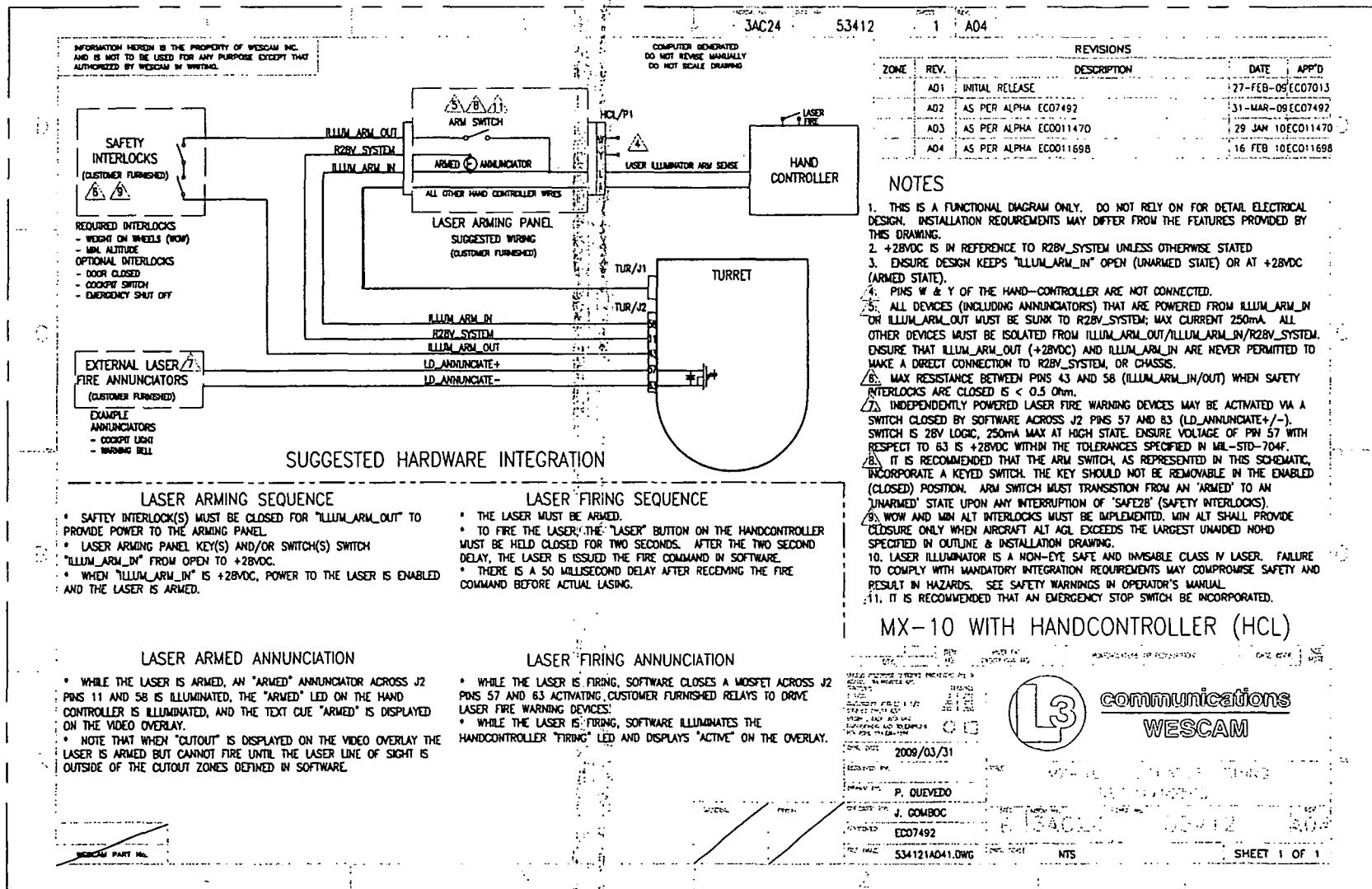
















Communications
WESCAM

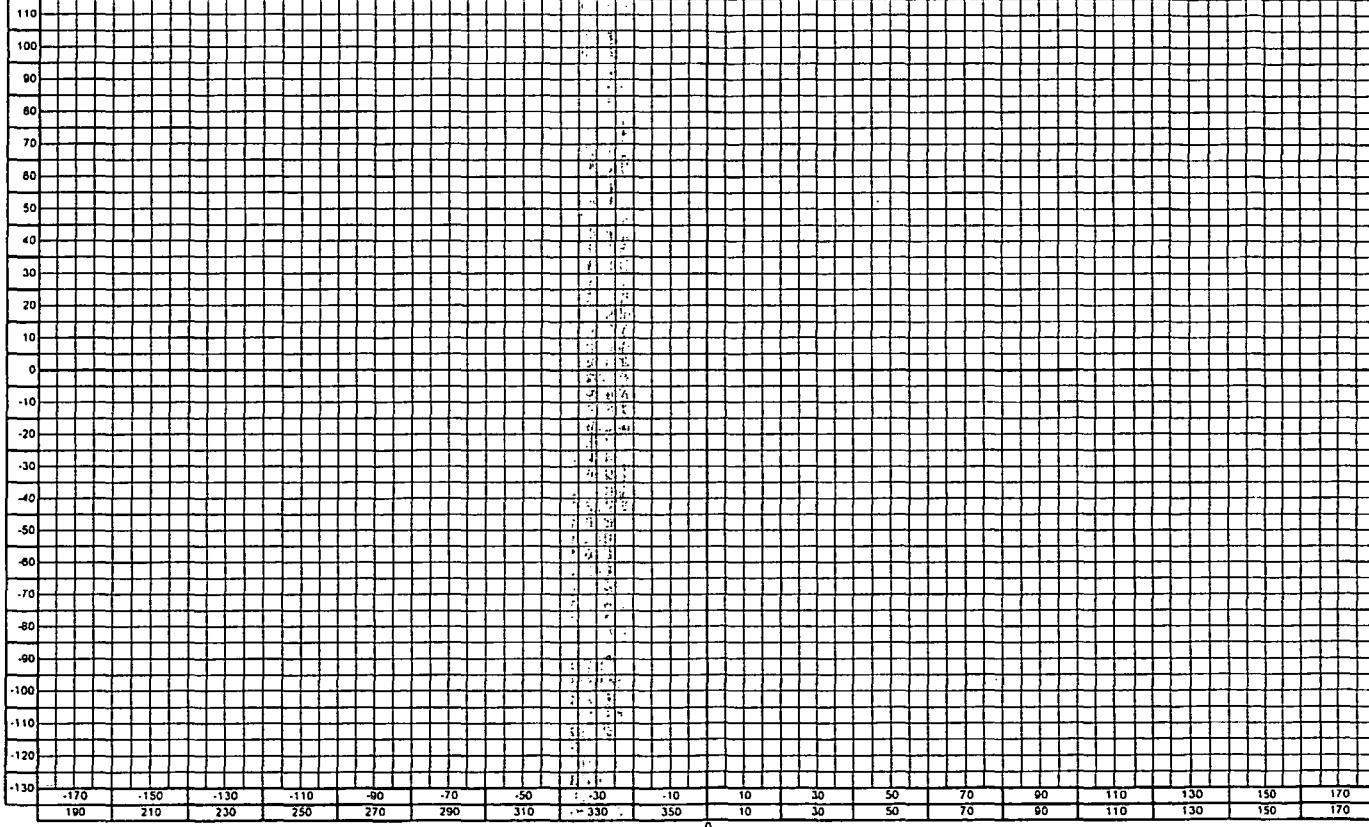
LASER CUTOUT DEFINITION FORM

FM1287
Rev.

Customer Name:

Date:

ELEVATION

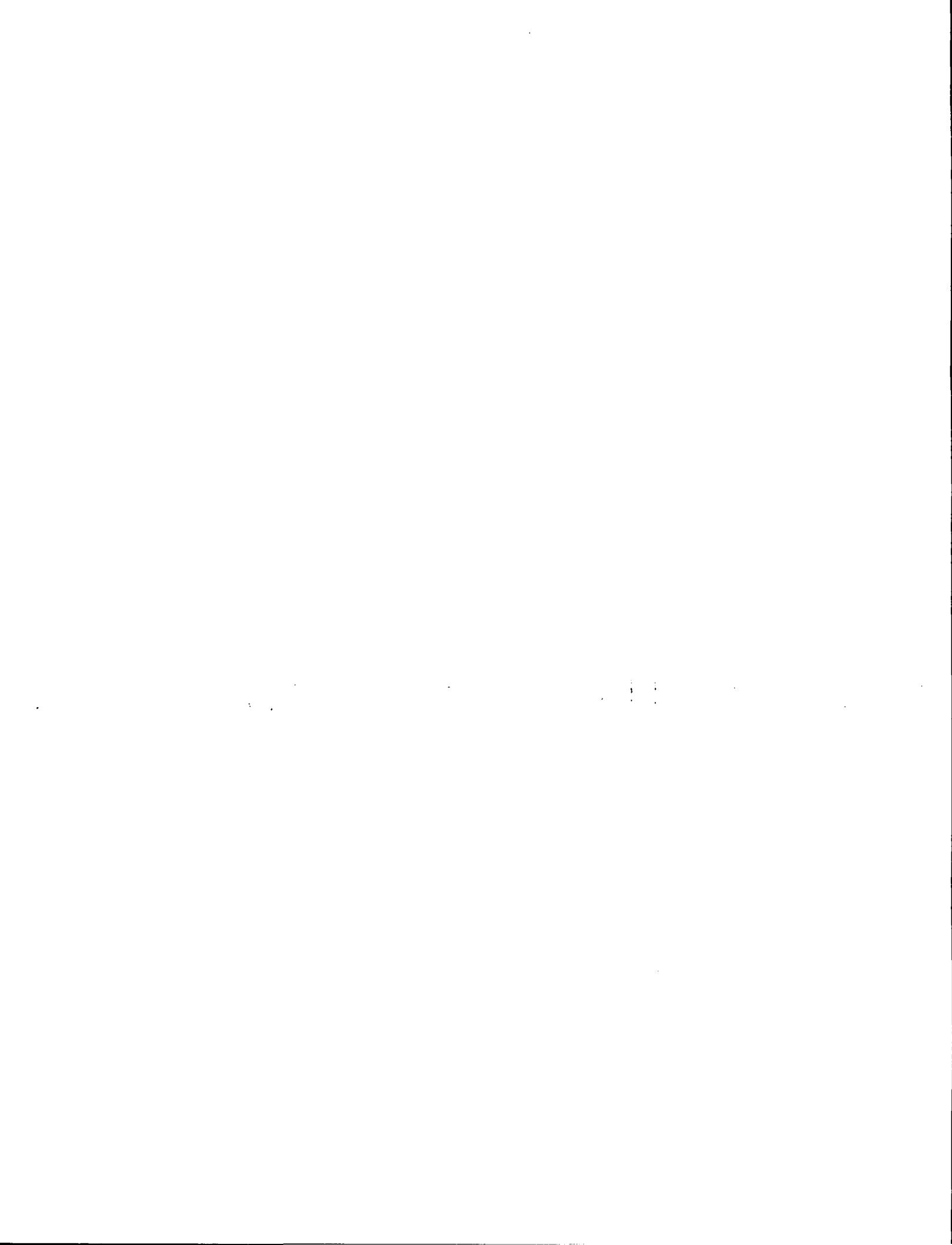


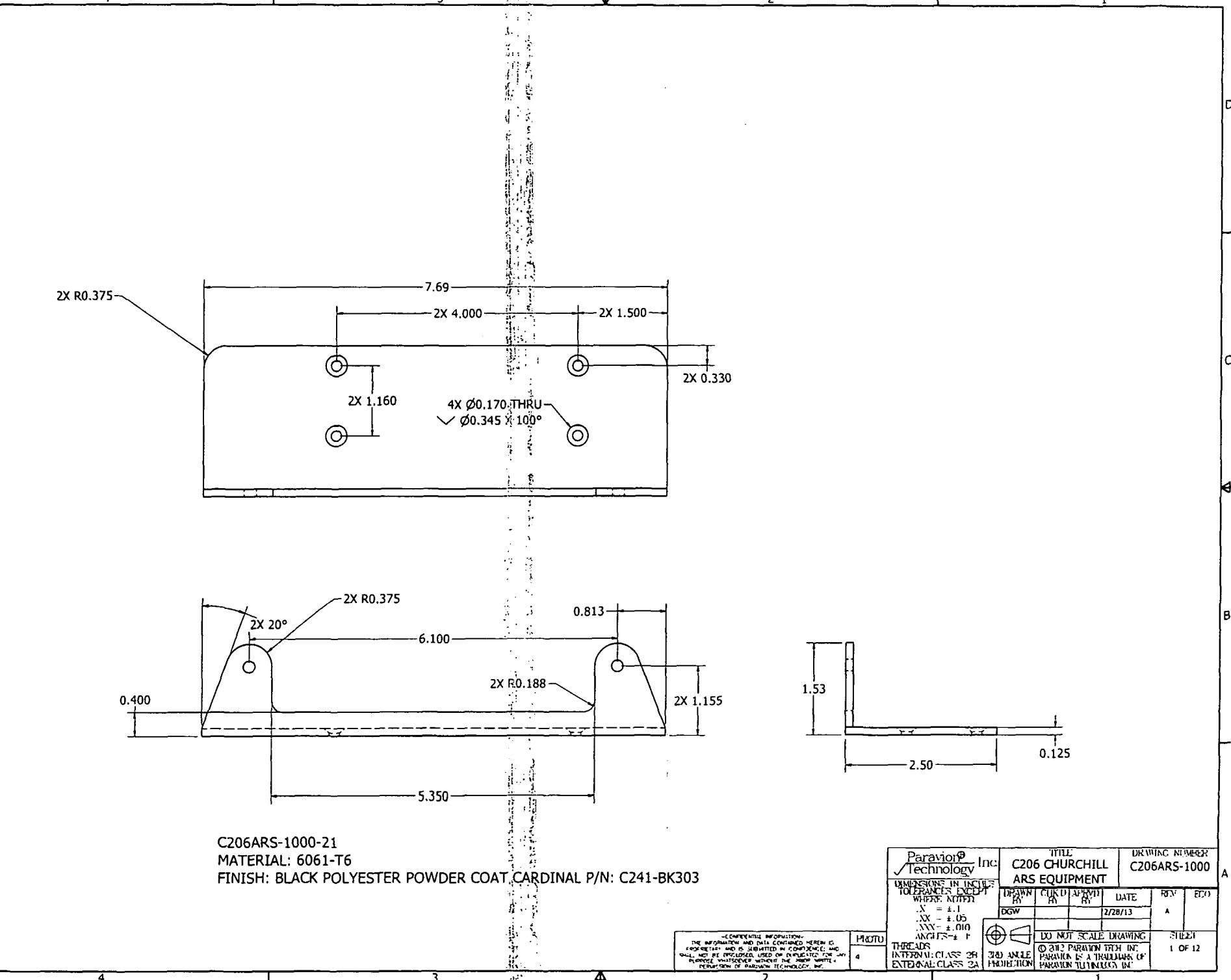
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Phone : 905-633-4000 Fax : 905-633-4100

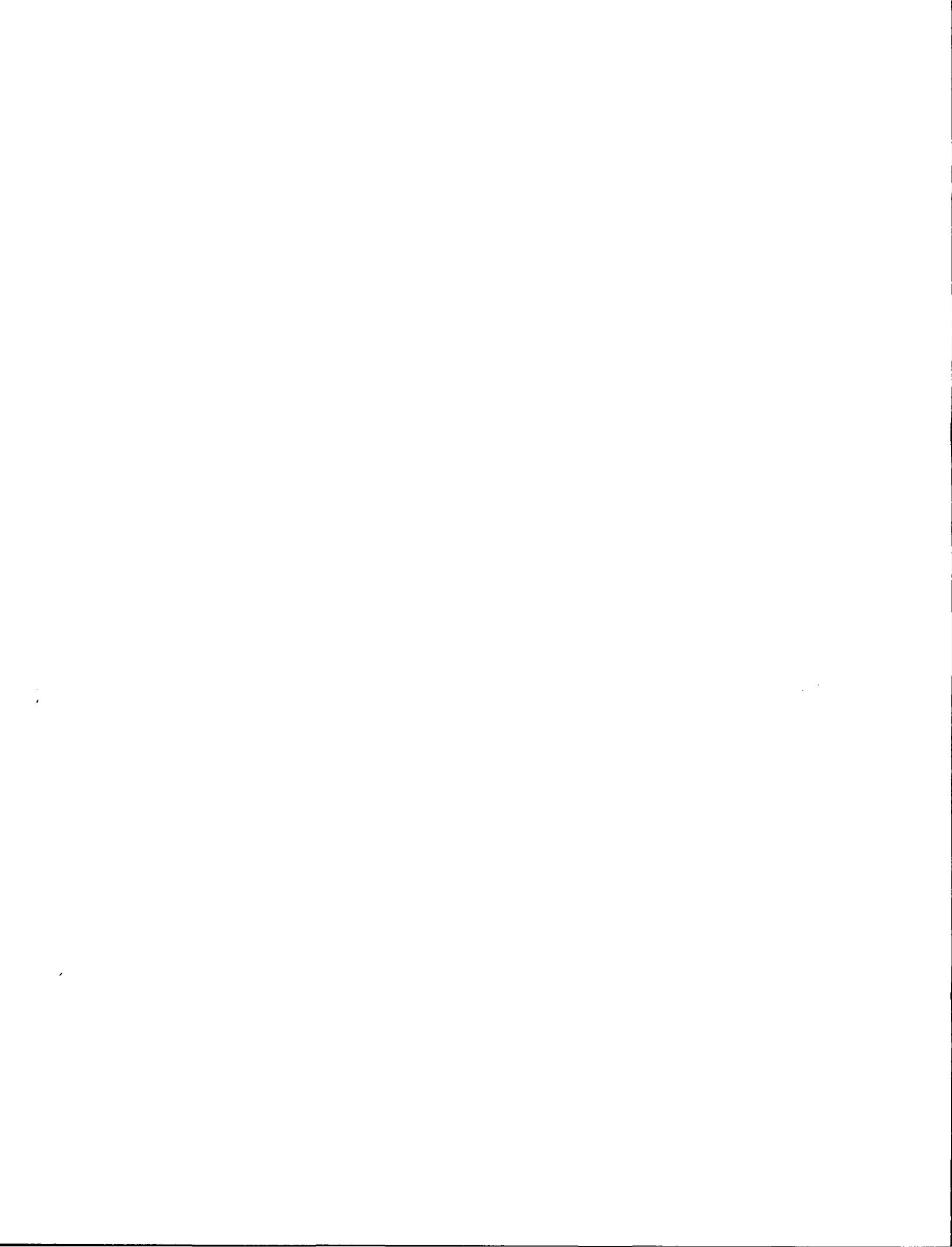
Sheet 1 of 2

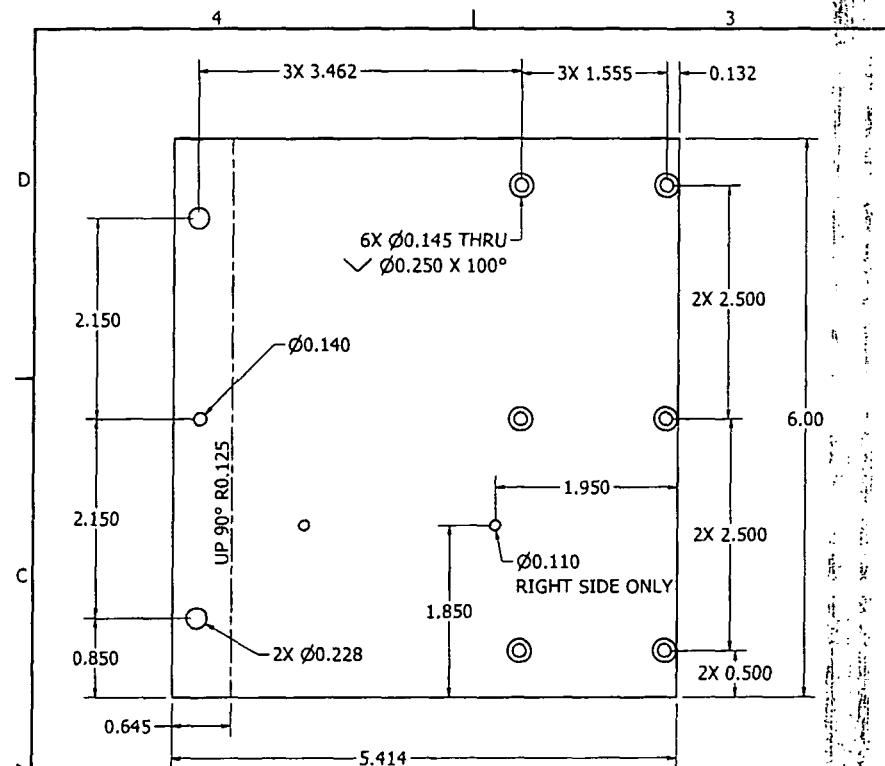
Approval:

Customer Signature

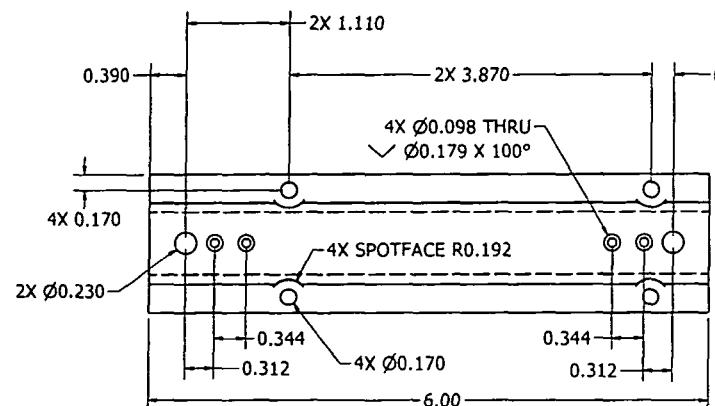




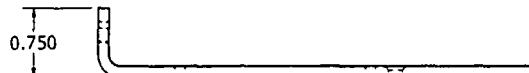
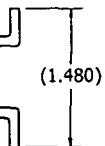




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FINISH: BLACK POLYESTER POWDER COAT CARDINAL
P/N: C241-BK303



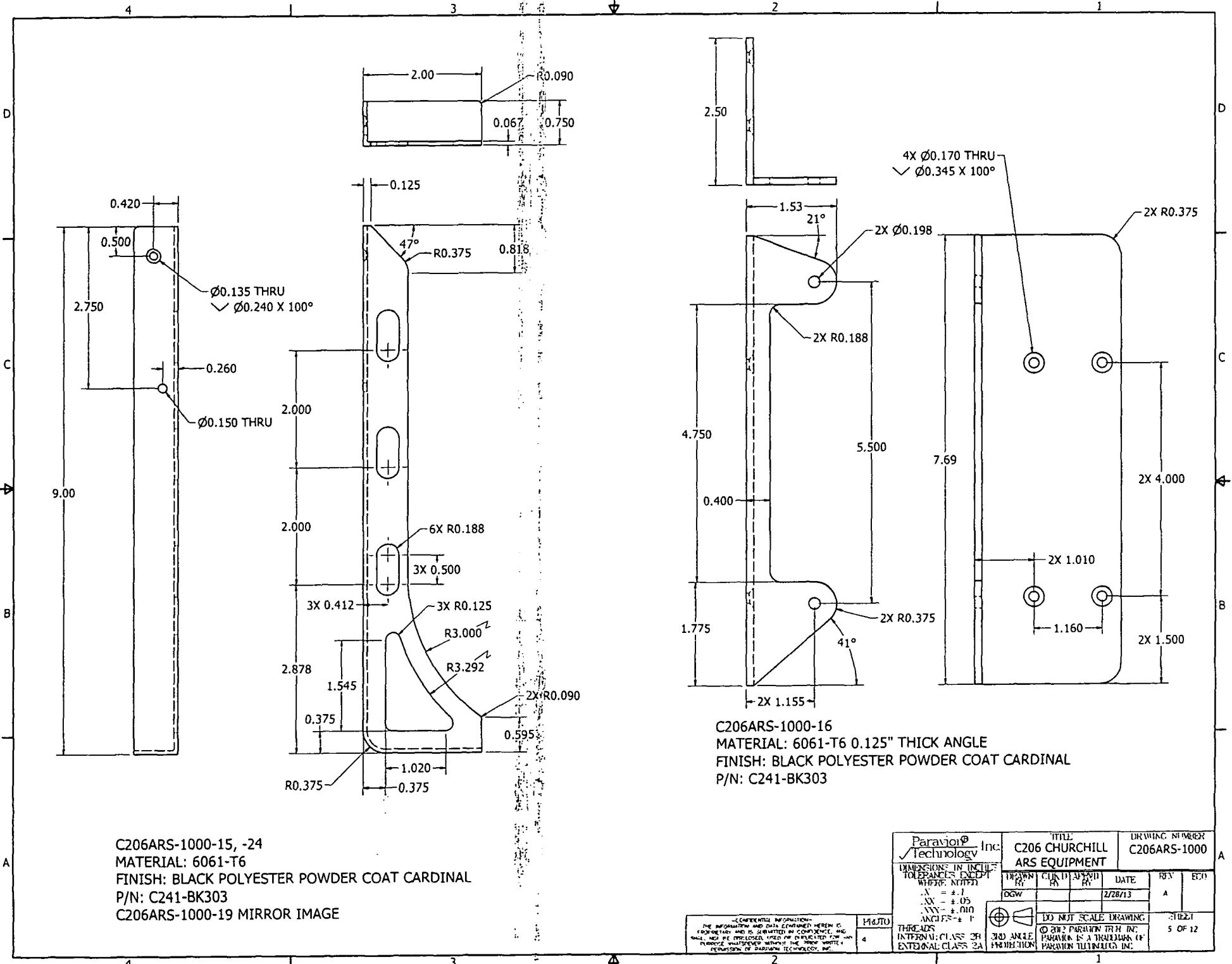
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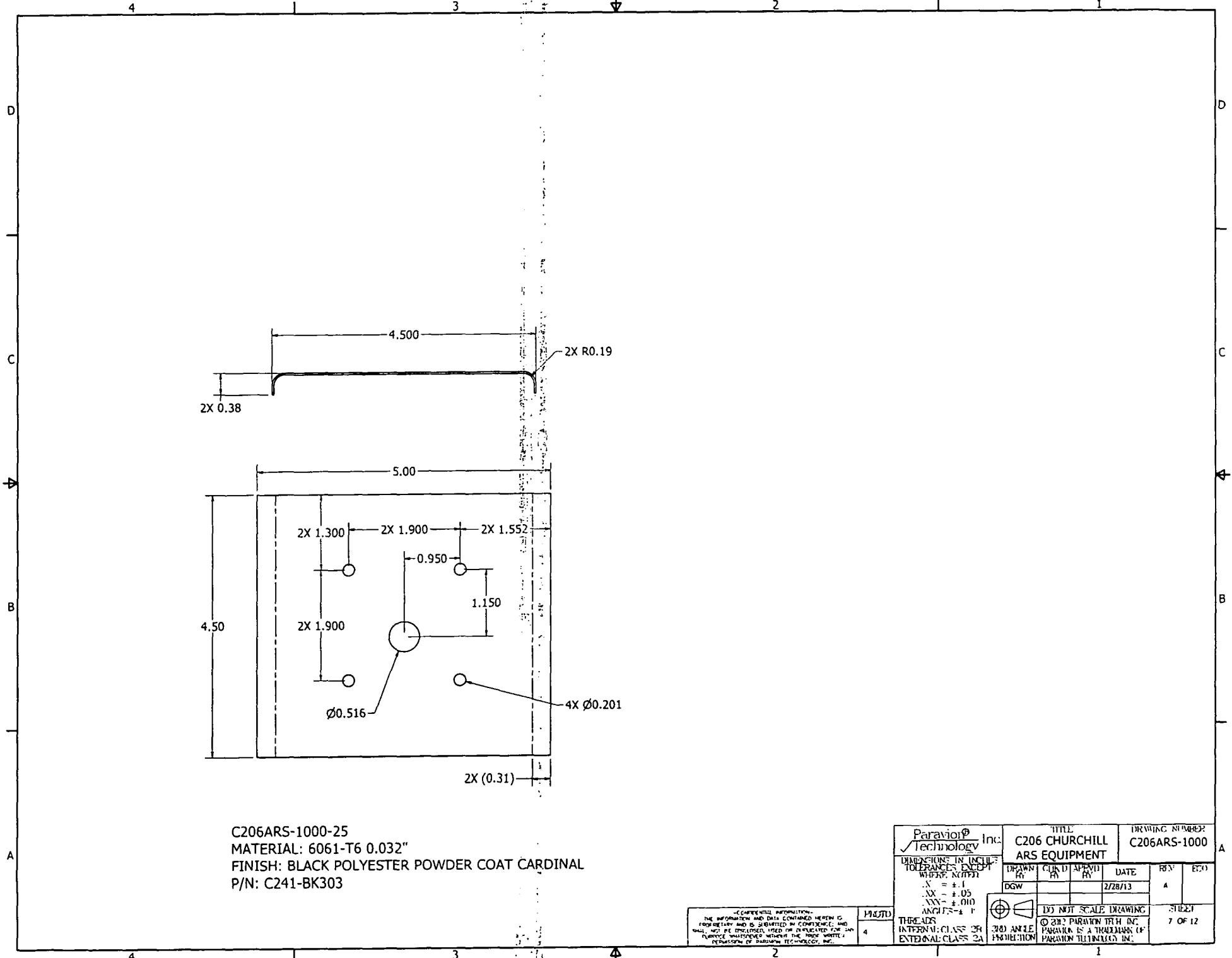


-CARTER
THE INFORMATION AND
CONFIDENTIAL AND IS
SHALL NOT BE DISCLOSED
PROMPTLY WITHDRAWN
REMISSION OF P

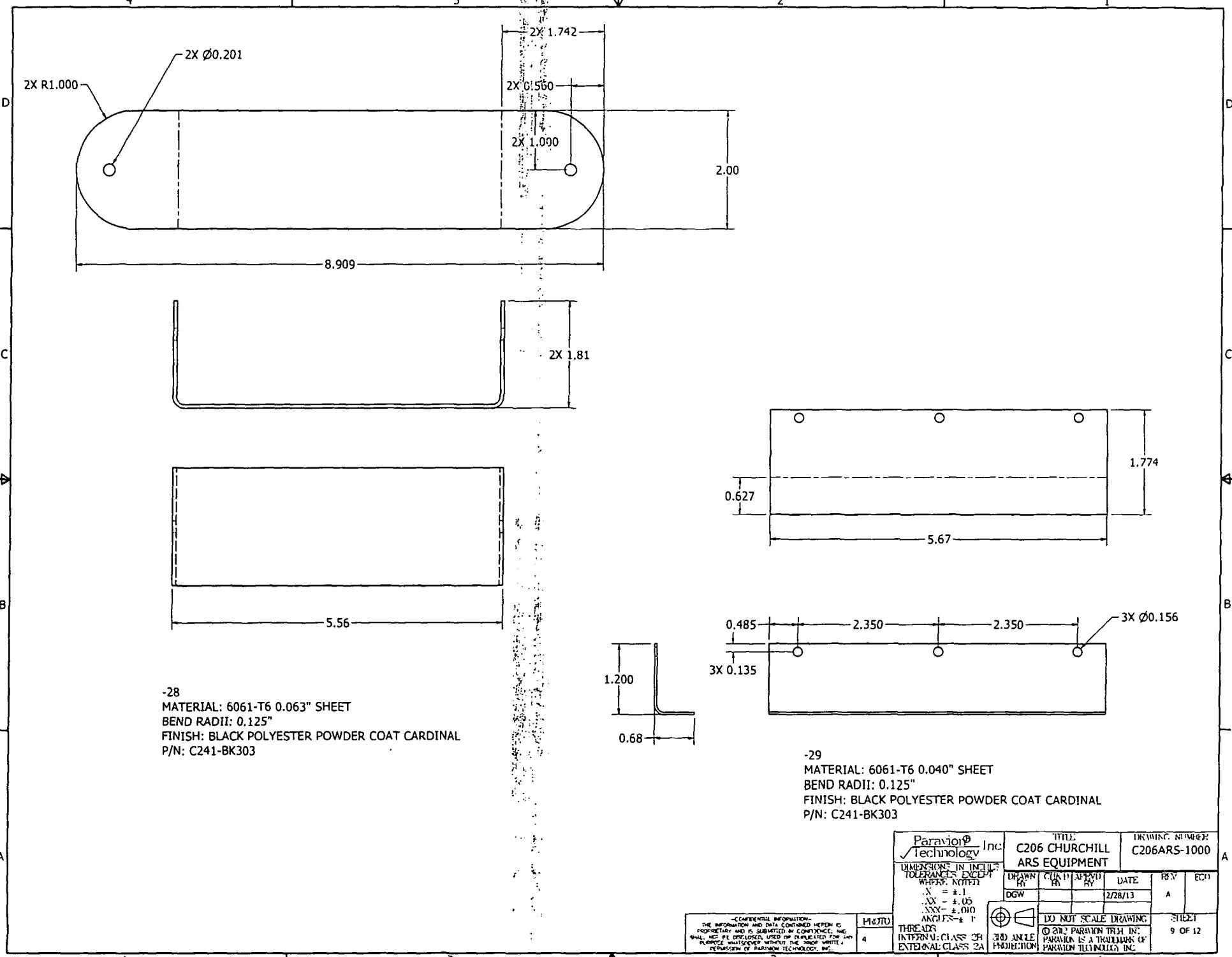
Paravisor [®] Technology Inc.		TITLE C206 CHURCHILL ARS EQUIPMENT		DRAWING NUMBER: C206ARS-1000	
DIMENSIONS IN INCHES TOLERANCES IN INCHES WHERE NOTED		DRAWN BY	CHECKED BY	APPROVED BY	DATE
.X = ± .1 .XX = ± .05 .XXX = ± .010 ANCHORS = ± 1		DGW			2/28/13
				REVISIONS A	
				ECC	
				STREET	
				3 OF 12	
THREADS INTERNAL CLASS 3A EXTERNAL CLASS 2A		BOD ANGLE PROHIBITION		DO NOT SCALE DRAWING © 2012 PARAVISOR TECH INC. PARAVISOR IS A TRADEMARK OF PARAVISOR TECHNOLOGY INC.	





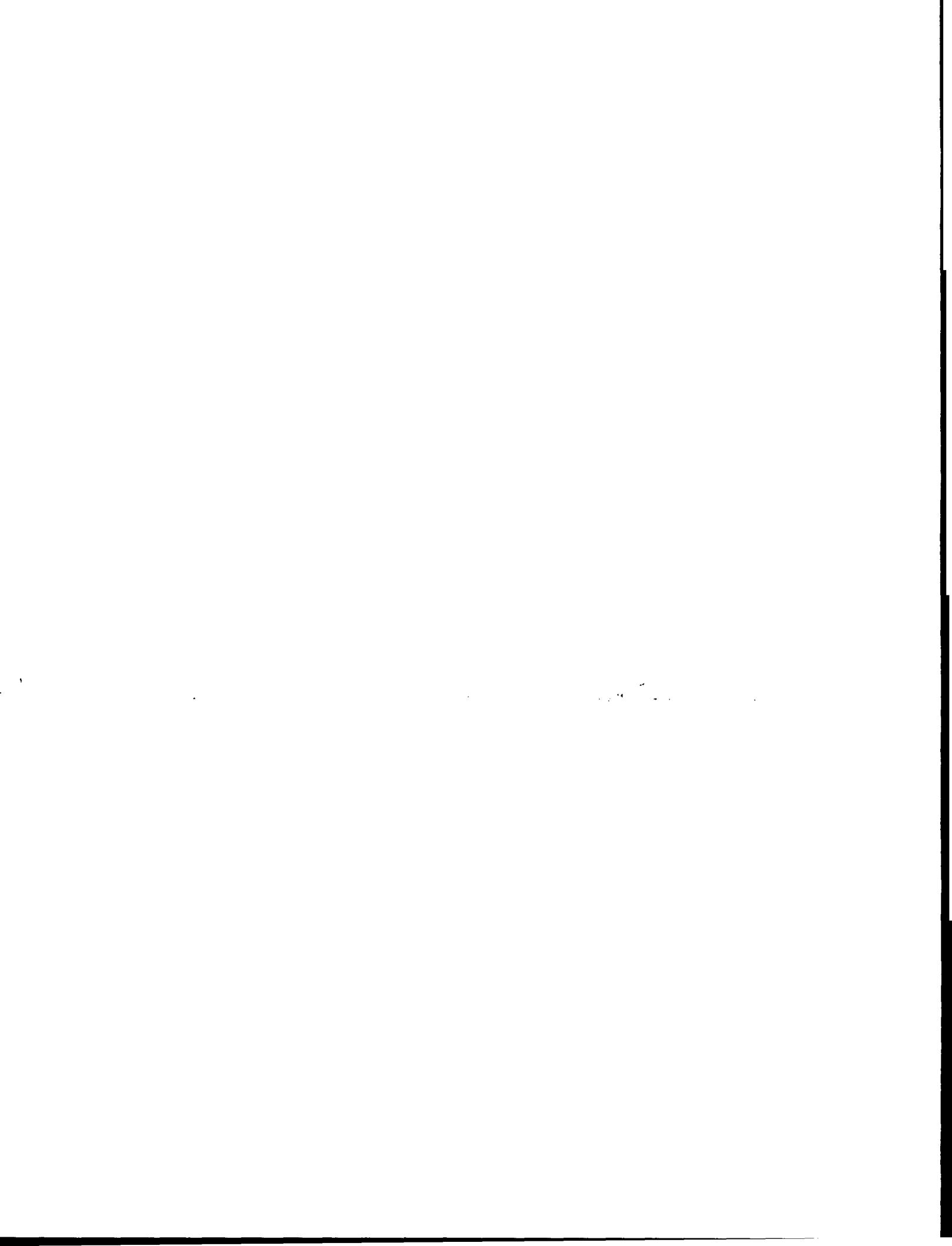






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Paravion® Inc. Technology		TITLE C206 CHURCHILL ARS EQUIPMENT		DRAWING NUMBER C206ARS-1000		
DIMENSIONS IN INCHES TOLERANCES EXCEPT WHERE NOTED: X = ± .1 XX = ± .05 XXX = ± .010 ANGLES = ± 1°		DRAWN BY DGW	CHANGED BY DGW	DATE 2/28/13	P.R.V. A	ECO
THREADS INTERNAL: CLASS 2B EXTERNAL: CLASS 2A	4	PROJ TO MATERIAL STAINLESS STEEL	30° ANGLE PROJECTION	1	9 OF 12	
DO NOT SCALE DRAWING © 2012 PARAVION TECH INC. PARAVION IS A TRADEMARK OF PARAVION TECHNOLOGY INC.	1					



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3

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APS
VICE01
IP

CAM

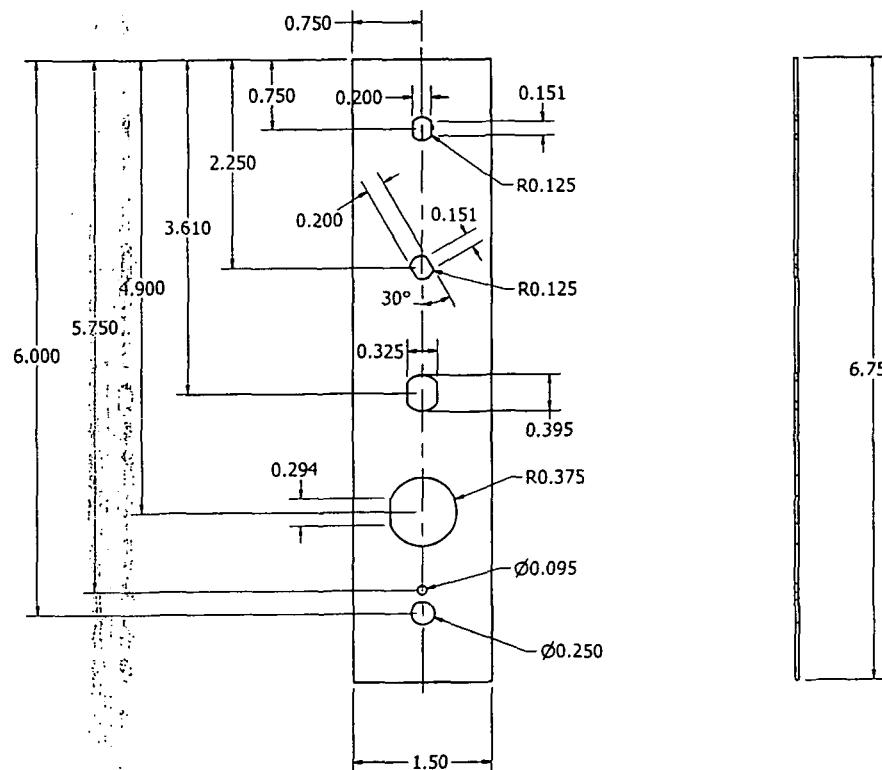
ARS
DLINK
VII-E:

BRIGHTNESS

MENU

ON

OFF



-31

MATERIAL: 6061-T6 0.040" SHEET
 FINISH: BLACK POLYESTER POWDER COAT CARDINAL
 P/N: C241-BK303

3

4

2

1

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PHOTO

4

Paravion® Inc. Technology		TITLE C206 CHURCHILL ARS EQUIPMENT		DRAWING NUMBER C206ARS-1000	
DIMENSIONS IN INCHES TOLERANCES: .005 WHERE NOTED X = ± .1 XX = ± .05 XXX = ± .010 ANGLES: ± 1°		DRAWN BY DGW	CHECKED BY HJM	APPROVED BY HJM	DATE 2/28/13
		REV. A	REV. B		
THREADS: INTERNAL CLASS 2A EXTERNAL CLASS 2A	30° ANGLE PROJECTION	DO NOT SCALE DRAWING © 2012 PARAVION TECH INC. PARAVION IS A TRADEMARK OF PARAVION TECHNOLOGY INC.	SHEET 11 OF 12		



Bill of Materials

*IR-440-1, REV. N/C

ITEM #	P/N	DESCRIPTION	QTY	TYP
0	*IR-440-1, REV. N/C	GIMBAL ADAPTER ASSY	1	KIT
1	IR-606-1	SUPPORT PLATE	4	EA
2	MS16998-44	BOLT	4	EA
3	NAS1149C0463R	WASHER	1	EA
4	MS24693S279	SCREW	1	EA
5	NAS43DD3-32FC	SPACER	1	EA
6	NAS1149F0332P	WASHER	1	EA
7	MS21042L3	NUT	1	EA



PARAVION TECHNOLOGY, INC	Instructions for Continued Airworthiness IR-605-1 Support Plate – Cessna 206 Revision: <u>IR</u> Date: <u>12/19/2012</u> A/C N#: _____ A/C S/N: <u>T20608983</u>
-------------------------------------	--

The installation is to be inspected in accordance with the following criteria or equivalent operator's Approved Airworthiness Inspection Program:

1.0 INTRODUCTION

These Instructions for Continued Airworthiness contain the necessary information for carrying out the ongoing maintenance and inspections on the installation of an IR-605-1 Support Plate on a Cessna 206 aircraft in accordance with FAA Form 337 dated _____.

2.0 DESCRIPTION

Paravion Technology drawing IR-605 describes the support plate used as provisions for a Wecam MX-10 camera. The plate is installed in the belly of the aircraft at approximately the center of the fuselage along the centerline of the aircraft. It weighs approximately 1.6 lbs and is installed using 4 x NAS1351C4 screws.

3.0 CONTROL, OPERATION INFORMATION

N/A

4.0 SERVICING INFORMATION

N/A

5.0 MAINTENANCE INSTRUCTIONS

The inspection program for this installation consists of a 12-month annual inspection for the condition of the support plate and associated components. This inspection is a complete visual inspection requiring only a single logbook entry.

12-Month Inspection

- A. Inspect condition of support plate and all associated mounting structure for loose hardware or damage, i.e. bent, cracked or dented structures, and repair or replace as necessary.

The 12-month inspections shall be accomplished by an appropriately rated mechanic assigned to this aircraft and can be accomplished earlier to match up with other aircraft inspections.

6.0 TROUBLESHOOTING

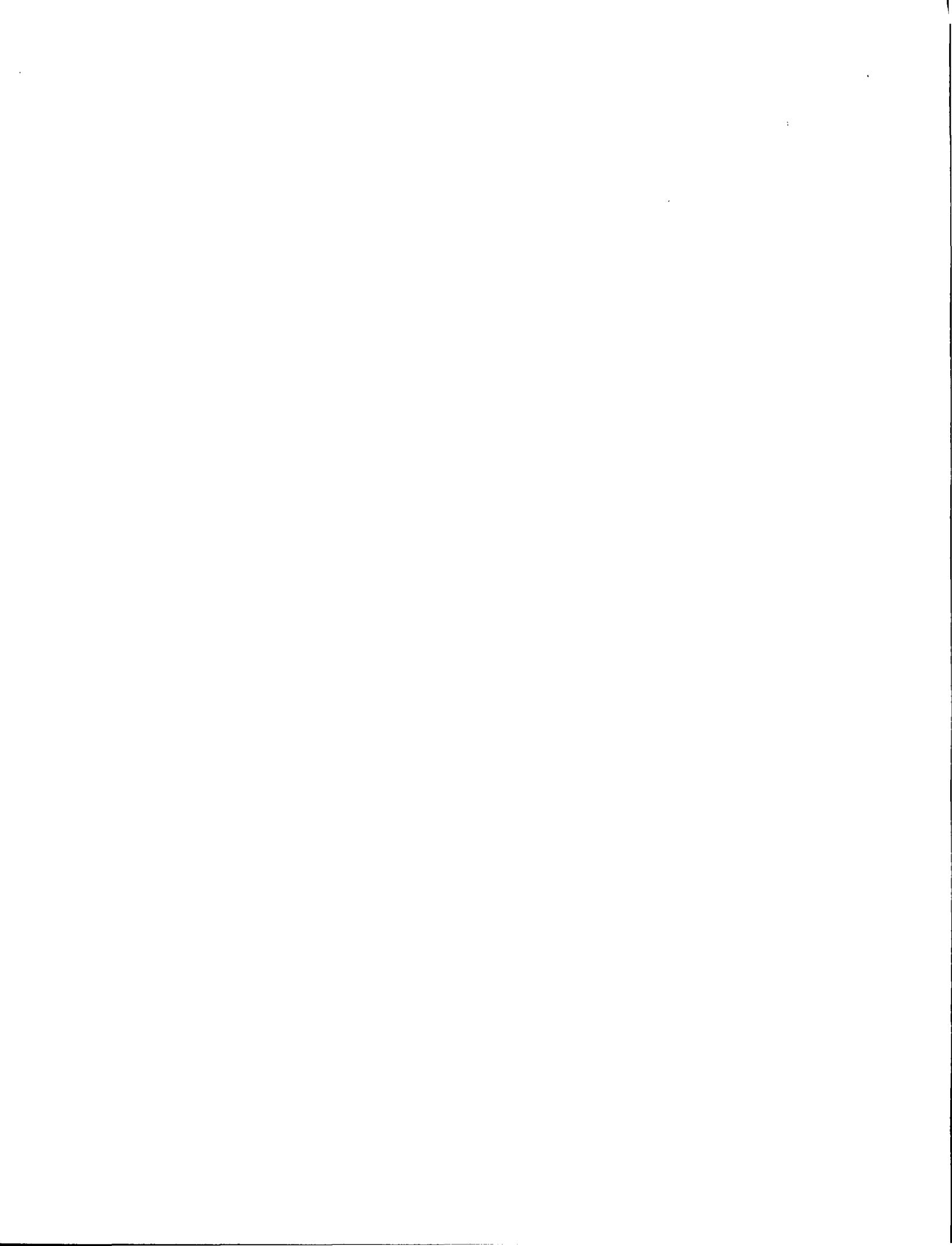
N/A

7.0 REMOVAL AND REPLACEMENT INSTRUCTIONS

- A. Paravion Technology drawing IR-605 (provided) shows the details of the support plate and report number ER-C206ELP-2 shows the installation of the plate and can be used as a reference in the event the plate needs to be removed and replaced.

8.0 DIAGRAMS

N/A





US Department
of Transportation
Federal Aviation
Administration

MAJOR REPAIR AND ALTERATION
(Airframe, Powerplant, Propeller, or Appliance)

Form Approved
OMB No. 2120-0020
2/28/2011

Electronic Tracking Number

For FAA Use Only

INSTRUCTIONS: Print or type all entries. See Title 14 CFR §43.9, Part 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. §44701). Failure to report can result in a civil penalty for each such violation. (49 U.S.C. §46301(a))

1. Aircraft	Nationality and Registration Mark N959JT	Serial No. T20608983	
	Make Cessna	Model T206	Series
2. Owner	Name (As shown on registration certificate) PSL Surveys	Address (As shown on registration certificate)	
		Address P.O. Box 756	City Bristow
		Zip 20136	Country USA

3. For FAA Use Only

The technical data identified herein has been found to comply with the applicable airworthiness requirements and is hereby approved for use only on the above described aircraft, subject to conformity inspection by a person authorized in CFR title 14, Part 43, section 43.7.

Approving Inspector: Juliette Sumner Date: 3/29/2013
Denver FSDO, NM-03

4. Type		5. Unit Identification			
Repair	Alteration	Unit	Make	Model	Serial No.
<input type="checkbox"/>	<input type="checkbox"/>	AIRFRAME	_____	(As described in Item 1 above)	_____
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT	_____	_____	_____
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER	_____	_____	_____
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type	_____	_____
			Manufacturer		

6. Conformity Statement

A. Agency's Name and Address		B. Kind of Agency		
Name Philip Glasgow		<input checked="" type="checkbox"/> U. S. Certificated Mechanic	Manufacturer	
Address 2533 Dallas Creek Court		<input type="checkbox"/> Foreign Certificated Mechanic	C. Certificate No.	
City Fort Collins State Co. CO		<input type="checkbox"/> Certificated Repair Station		
Zip 80528 Country USA		<input type="checkbox"/> Certificated Maintenance Organization	A&P 3292572 IA	

D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

Extended range fuel per 14 CFR Part 43 App. B	<input type="checkbox"/> Signature/Date of Authorized Individual Philip Glasgow 4/2/13
---	--

7. Approval for Return to Service

Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is Approved Rejected

BY	FAA Fit. Standards Inspector	Manufacturer	Maintenance Organization	Persons Approved by Canadian Department of Transport
	FAA Designee	Repair Station	<input checked="" type="checkbox"/> Inspection Authorization	Other (Specify)

Certificate or Designation No. A&P 3292572 IA	Signature/Date of Authorized Individual Philip Glasgow 4/2/13
--	---

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N959JT

4/2/13

Nationality and Registration Mark

Date

-Installed a Paravion Technology Inc Infared camera mounting provisions IAW STC STC SA 00295DE for a L3 Wescam MX10 camera system.

-Installed a Churchill Augmented Reality System IAW manufacturers installation drawings # ARS 500C-201210 Rev 6 10/24/12. Power is supplied from the avionics buss and is protected using a Klixon C/B P/N 7277-2-3 labeled "Mapping" Mounted the ARS system to the above installed Infrared camera mount and secured the ARS to the mount by fabricating 2 X support brackets. Ref Paravion Technology drawing 206ARS-1000 sheet 1 for full fabrication details. Mounted the GPS antenna to the roof of the aircraft structure at station 104.0 using manufacturer provided hardware.

-Mounted a 9.0"Airborne display monitor into the instrument panel. The primary display monitor is mounted to the instrument panel on the R/H side using 2X MS24693-363 screws. Attached 2X MS21059-L3 nut plates to the instrument panel using 4 X MS21426-3-4 countersunk rivets. The remote control unit is provided power from the avionics buss and is protected using a 3 Amp Klixon C/B P/N 7277-2-3. And is labeled "Monitor ". The remote control unit for the monitor is mounted to aircraft structure behind the instrument panel. Attached the control unit to the support brackets using 4 X screws P/N MS24693S26 and 4 X clip nuts P/N 294667. Fabricated the two supports from stock 6061 T6 aluminium and machined the support brackets. Ref Paravion Technology drawing 206ARS-1000 sheet 5 for fabrication details. Attached the support brackets to the instrument panel using 4 X screws P/N and 4 X nutplates P/N which are riveted to the support brackets using 8 X MS20426AD3-5 rivets. Fabricated a support brace from 6061 T6 aluminium 0.063" and bent to a 90 deg angle. Attached the brace to the supports using 2 X nutplates P/N MS21059L06 and 2 X screws P/N MS24693S26. Fabricated a plate for the remote control controls from 6061 T6 aluminum 1.5" X 4". Secured the power switch, dimmer switch, menu control switch, Video selection switch & the Downlink switch to this panel using the manufacturers provided switches. Secured the panel to the arm rest of the interior plastic using 4 X MS35206-226 screws, 4 X AN 960-6L washers & 4 X MS21083N06 nuts.

-Mounted The Jantec Downlink Control ECU to the floor at station 133.75 using 4 X MS27039-1-09 screws. Attached 3 X nut plates P/N MS2105L3N and attached to the existing structure using 6 X CR3213 4-4 rivets. Fabricated a doubler from 6061 T6 .063" 8.5" X .7 X.7 angle. Attached two of the afore mentioned nut plates to this doubler using the rivets mentioned above. Installed the Jantec Down link IAW manufacturers Dwgs . System is protected using a Klixon C/B switch P/N 7270-3-10 and is labeled "Down Link" Mounted two antennas on the bottom of the aircraft. Mounted the first antenna at station 150.0" on the bottom of the aircraft to the R/H side of the aircraft center line. Fabricated a doubler from 6061 T6 aluminium 4" X 5". Attached the antenna to the aircraft using 4 X P/N MS51987-48 screws, 4 X P/N AN960C8 washers & 4 X P/N MS21042-L08 nuts. Mounted the second antenna to the bottom of the aircraft at station 159.0" to the L/H side of the center line. Fabricated a doubler from 6061 T6 aluminium 4" X 5". Attached the antenna to the aircraft using 4 X P/N MS51987-48 screws, 4 X AN960C8 P/N washers & 4 X P/N P/N MS21042-L08 screws. Mounted the control head to the center console using 4 X P/N 2-56 screws. Fabricated a double and machined to fit. ref Paravion Technology Dwg C206ARS-1000 sheet 11 for fabrication details.

-Installed 2 X Aux Foot switches on the floor at station location 20.00". Fabricated foot switch holder form the same material as mentioned above for the center console and installed a 2 X switches P/N M8805/55-001 X 2. Attached the Foot switch housing Using 2 x MS35206-228 screws and 2 X AN960JD6L washers. to the floor using 3 X Nut plates P/N MS21075L06 & 1 X MS21069L06 nut plate. Attached the nut plates to the floor using 8 X MS20426AD3-3.5 Rivets.

Additional Sheets Are Attached

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N959JT

4/2/13

Nationality and Registration Mark

Date

-Fabricated a breaker panel from .25" X 6.5" X 6.5" stock. Machined the required holes and location for the various connectors required. fabricated 2 X angles form .040" 6.5"X .75" X 1.25" 6061 T6 alüminium. Attached the angles to the breaker panel using 6 X screws P/N MS24693S26. Attached the assembly to the airframe using 4 X screws P/N MS24693S26. Attached 4 X nut plates P/N MS21069L06 to the existing structure using 8 X MS20426AD-3-4 rivets. Ref Paravion Technology Dwg C206ARS-1000 sheet 8 for fabrication details.

- Fabricated a carbon fiber housing to mount 2 X USB ports and 1 X hand controller cannon plug. Attached the housing to the aircraft structure on the floor between the seats at station 55.0 just aft of the existing vent using 4 X MS21075L3N nut plates. Attached the nut plates to the the floor using 4 X MS20426-3-4 countersunk rivets. used 4 X screws P/N MS27039-1-09 screws and 4 X AN960C10L washers. Ref attached Paravion Technology dwg C206ARS-1000 sheet 10 for fabrication details of the housing.

- Fabricated a mount for the existing Motorola XTVA radio housing form .063 6061 T6 aluminium 9" X 2". Attached the mount to the floor aft of the USB housing at station 65.0 using 2 X MS21075L3N nut plates. Attached the nut plates to the the floor using 4 X MS20426-3-4 countersunk rivets. used 2 X screws P/N MS27039-1-09 screws and 2 X AN960C10L washers. Ref Paravion Technology drawing C206ARS-1000 sheet 9 for fabrication details.

Wire gauge selection was done in accordance with AC43-13-1B Chapter 11, Aircraft Electrical System, section 5 (wiring rating) paragraphs 11-66, 11-67 section 6 (Aircraft Electrical Wire section) paragraphs 11-76, 11-77.

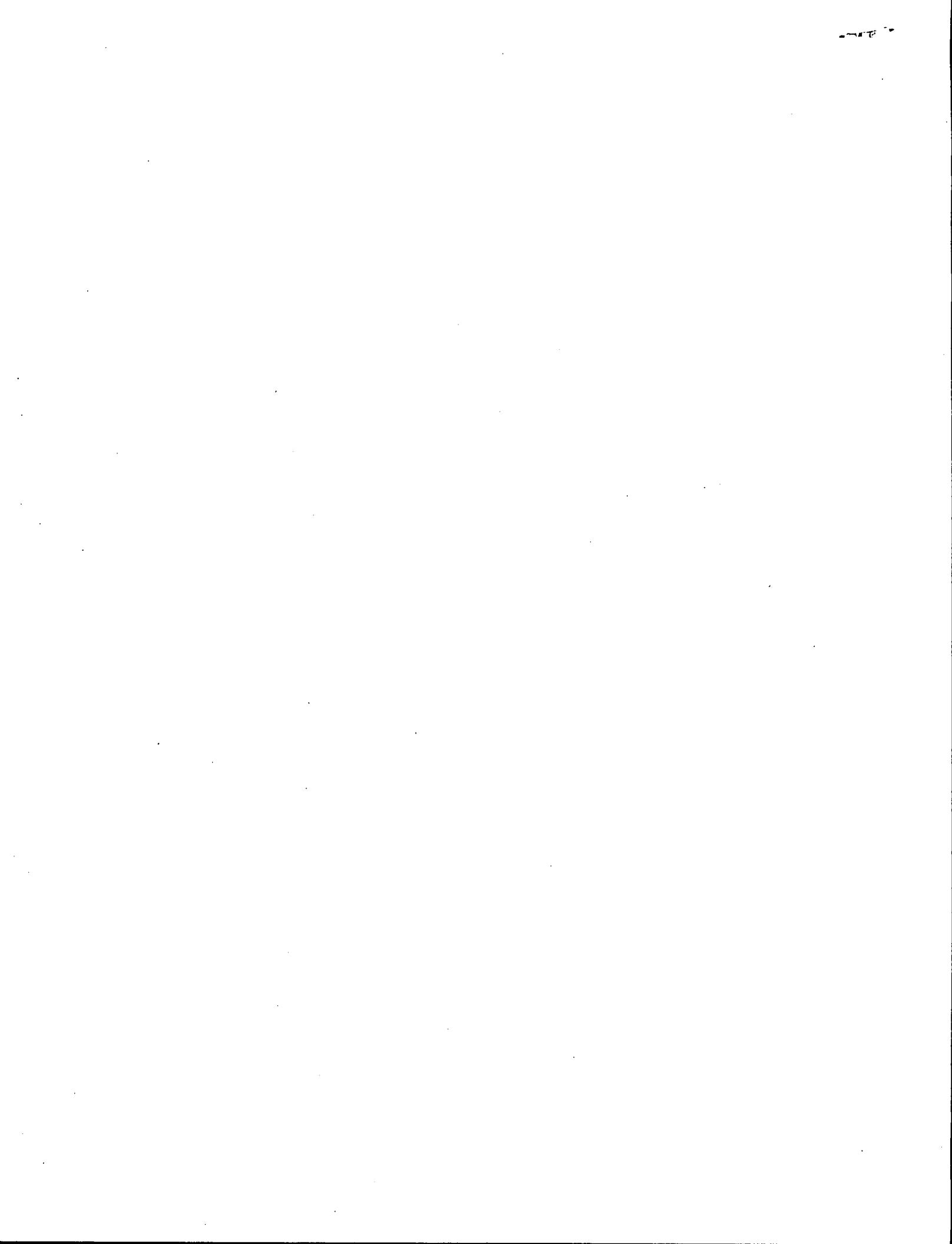
An electrical load does not exceed limitations of AC43-13-1b Chapter 11, paragraphs 424 (Electrical load limits), 425 (generator) and 428 (determination of electrical load).

The Instructions for Continued Airworthiness (ICA) contained in the Flight Standards Handbook Bulletin for Airworthiness (HBAW-8900.1) are not applicable as these components are not field repairable and are "Remove and Replace" items only.

Aircraft weight & balance and equipment list amended as required.

----- Nothing follows -----

Additional Sheets Are Attached



United States of America
Department of Transportation—Federal Aviation Administration

Supplemental Type Certificate

Number SA00295DE

This certificate, issued to
Paravion Technology, Inc.
2001 Airway Avenue
Fort Collins, CO 80524

certifies that the change in the type design for the following product with the limitations and conditions therefor as specified hereon meets the airworthiness requirements of Part 3 of the Civil Air Regulations.

Original Product—Type Certificate Number: A4CE

Make: Cessna Aircraft Company
Model: TU206G, 206H, & T206H

Description of the Type Design Change:

Installation of an external Infrared Imaging System in accordance with Paravion Technology Master Drawing List Report No. DL-C206IR-100, Revision N/C, dated March 29, 1997 or later FAA approved revision.

Limitations and Conditions:

1. This approval should not be extended to aircraft of this model on which other previously approved modifications are incorporated unless it is determined that the interrelationship between this change and any of those other previously approved modifications, including changes in type design, will introduce no adverse effect upon the airworthiness of that aircraft.
2. A copy of this Certificate and Flight Manual Supplement must be maintained as part of the permanent records for the modified aircraft.
3. FAA approved Aircraft Flight Manual Supplement PR-C206IR-100M, Revision 0, dated June 11, 1997 or later FAA approved revision is required.
4. If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission.

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.

Date of application: January 10, 1997 Date reissued:

Date of issuance: June 12, 1997 Date amended: April 8, 2004



By direction of the Administrator

A handwritten signature in black ink that reads "Melissa Sandow".

Melissa Sandow, (Signature) Small Airplane Program Manager

Northwest Mountain Region

Denver Aircraft Certification Office

(Title)

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

This certificate may be transferred in accordance with FAR 21.47.



**PARAVION TECHNOLOGY, INC.
2001 AIRWAY AVENUE
FT. COLLINS, COLORADO 80524**

REPORT NO. PR-C206IR-900M

INSTALLATION INSTRUCTIONS

FOR

INFRARED IMAGING SYSTEM



REVISIONS

<u>REV.</u>	<u>DATE</u>	<u>DESCRIPTION</u>	<u>BY</u>
N/C	11/02/00	Original	MR
A	05/18/01	Added Video Output Note.	MR
B	09/06/02	Added reference to C206IR-101-2 Support Installation, section 2.1.1.	GP
C	10/25/04	Section 2.1.1 added reference to FLIR U8000, U8500 Section 2.1.3 re-worded to clarify doubler installation Added Table 2.2, other minor wording changes to clarify	REB
D	01-07-05	Section 2.2.10 edited to include assembly of Item 33 Doubler and Item 26 beam Assembly.	REB
E	12/06/05	Sect. 2.1.8, page 1 was "... Remove fasteners which conflict with angle installation. Adjust clamps to support tube in center of opening and level tube to cabin floor." Clarified to indicate positioning laterally and longitudinally.	REB



TABLE OF CONTENTS

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1.0 Introduction	1
2.0 Installation Procedures	1



1.0 INTRODUCTION

This document provides a step-by-step procedure for installation of the C206IR-100 Infrared Camera System Installation in the Cessna 206 Aircraft. The instructions contained herein are intended to supplement the information contained on the installation drawings.

2.0 INSTALLATION PROCEDURES

2.1 Support Installation (Drawing C206IR-101)

- 2.1.1 If your camera system has a dual power/control cable from the gimbal to the electronic control unit (Ref. FLIR MK-I and MK-II) then use C206IR-101-1. If your camera system has a single power/control cable from the gimbal to the electronic control unit (Ref. FLIR MK-III, U7000 and U7500, U8000, U8500) then use C206IR-101-2.
- 2.1.2 Remove baggage floor covering and all necessary side panels.
- 2.1.3 Verify location of doubler between longitudinal stringers in baggage compartment area and trim doubler/shim as necessary. Mark location of doubler on fuselage and remove all conflicting fasteners. Match drill doubler to existing fastener holes. Locate and drill additional fastener holes per drawing. Remove doubler, de-burr holes and install using indicated hardware.
- 2.1.4 Locate and drill indicated hole through both fuselage and doubler as shown, de-burr. Install rivets around hole through fuselage and doubler.
- 2.1.5 Adhere extrusion to circumference of opening.
- 2.1.6 Temporarily clamp angles to support assembly.
- 2.1.7 Position clamped support assembly in aircraft through hole and perpendicular to aircraft centerline.
- 2.1.8 Remove fasteners which conflict with angle installation. Adjust clamps to center the tube in the previously drilled opening and parallel to the baggage compartment floor (laterally). The support tube longitudinal angle should be set by leveling the Electronic Control Unit Mount Bracket to the cabin floor, not to the baggage compartment floor.
CAUTION: The C206IR-2500-1 Spacer (if used) and IR-1030-1 Angle are not symmetrical. Note correct orientation of parts before drilling baggage compartment floor.
- 2.1.9 Mark and match drill floor to support angles, remove clamped assembly.
- 2.1.10 Temporarily install indicated beam Assembly and support angles as shown. Match mark the beam for angles installation. Attach the angles to the beam in accordance to the drawing and temporarily re-install the assembly. Match drill the Beam Assembly to the previously drilled floor. Remove the beam Assembly and install the indicated Doubler (Nut Plate Assembly) using indicated fasteners(**NOTE:** It will be necessary to



trim the width of the doubler to fit inside the beam). Permanently install the beam Assembly using indicated hardware.

- 2.1.11 Match drill support legs to angles and secure using indicated hardware.
- 2.1.12 Reinstall support assembly by securing angles through spacer(optional if needed to adjust height) into the installed fastener assemblies in floor using indicated hardware.
- 2.1.13 Reinstall floor covering and fairing, trimming as necessary.
- 2.1.14 Install placard in a conspicuous location near existing baggage weight limits placard.
- 2.1.15 Optional use of MIL-S-8802F Class B2 sealant and DC4 or equivalent products, as indicated, may be desirable.

2.1 **Equipment Cabinet Installation (Drawing C206IR-201)**

- 2.2.1 Aircraft built prior to 1997 incorporate a lighter seat rail and require use of the C206IR-201-1 installation. The heavier seat rails in post-1997 aircraft require use of the C206IR-201-2 installation.
- 2.2.2 The equipment cabinet mount plate assembly may be installed to the seat rail pair in place of the copilot seat.

NOTE: See Table 2.2 for available Mount Plate options

TABLE 2.2; EQUIPMENT CABINET MOUNT PLATE INSTALLATIONS

Aircraft Mfr. Date	C206IR-201-1 Equipment Cabinet Installation	C206IR-201-2 Equipment Cabinet Installation
Pre-1997	C182IR-2500-1 Mount Plate Assy. Optional C182IR-2500-3 Assy.	
1997 and Later		C182IR-2500-2 Mount Plate Assy. Optional C182IR-2500-4 Assy.

- 2.2.3 Install Mount Plate Assembly to rail pair in desired location by sliding clamps onto rails. Mark locations for seat pin assemblies and remove to drill indicated holes.
- 2.2.4 Reinstall mount plate assembly to rail, slide clamps tight against rails and tighten screws.
- 2.2.5 Secure FWD/AFT movement by installing seat pin assemblies in drilled holes.
- 2.2.6 Install cabinet by inserting studs on bottom of cabinet into slots in the Mount Plate assembly and slide forward to small end of slot. Secure by inserting bolt through Mount Plate assembly and into cabinet nut plate.
- 2.2.7 Assure all fasteners are securely installed.



NOTE: Weight and balance data must be adjusted in accordance with actual weights and locations of installed equipment.

2.2 Equipment Installation (Drawing C206IR-251)

NOTE: All video outputs to any monitors should come from the VCR if installed.

The monitor and electronics support module (EU) installations are addressed by this drawing.

The equipment cabinet installation is designed to carry up to 25 lb. The equipment mounting bracket for the electronics support module is provided on the C206IR-1010-2 support assembly.

2.3.1 Monitor Installation:

2.3.1.1 The swivel support is designed to carry the Inframetrics monitor. Other monitor installations may require different mounting provisions and separate approval.

2.3.1.2 Remove the top cover of the monitor, then drill and install plate nuts on each side using the indicated rivets.

NOTE: Cover monitor assembly when modifying top cover to keep foreign material out. Reinstall monitor cover.

2.3.1.3 Adhere extrusion to perimeter of monitor glare shield using indicated adhesive.

2.3.1.4 Using indicated hardware, install the swivel support assembly to the monitor.

2.3.1.5 Place the monitor assembly on the top of the equipment cabinet, and secure using indicated hardware.

2.3.2 Electronics Unit Installation:

2.3.2.1 Install IR-2000-1 buttons to EU using indicated hardware.

2.3.2.2 Remove the clip from the rail assembly of EU mount.

2.3.2.3 Move the sliding shafts away from the keyholes in the rail assembly.

2.3.2.4 Fit the buttons on the EU into the keyways, and slide them into the keyway slots.



2.3.2.5 Move the sliding shafts against the EU and tighten the lock knobs. Replace the clip.

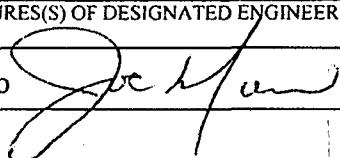
NOTE: Weight and balance calculations must include installed equipment.

2.4 Power Supply Wiring (Drawing C206IR-301)

- 2.4.1 Refer to drawing and camera system specification data for cable identification and connection.
- 2.4.2 Remove panels as necessary.
- 2.4.3 Install indicated circuit breaker in available aircraft breaker panel position. Provide electrical power through avionics buss.
- 2.4.4 Locate unused rocker type switch in lighting panel for use as infrared on/off switch.
- 2.4.5 Route power cable to electronics unit from infrared on/off switch. General cable routing should follow existing electrical wiring.
- 2.4.6 When system is installed for use, loose cables should be routed under seats and otherwise secured.
- 2.4.7 When system is disabled or removed, loose cables and controls must be stowed or removed.



SEQ No. 12JM003PA

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS				1. DATE 12/28/2012
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION				
2. MAKE Cessna Aircraft Company	3. MODEL NO. 206	4. TYPE (<i>Airplane, Radio, Helicopter, etc.</i>) Airplane	5. NAME OF APPLICANT Paravion Technology Inc.	
LIST OF DATA				
6. IDENTIFICATION		7. TITLE		
Paravion Technology Drawing IR-605 Rev A dated 11/13/2012 Paravion Technology Document ER-C206ELP-2 (MX-10 Installation) Rev 4 dated 12/21/2012		Support Plate Structural Analysis for Wescam MX-10 Installation Using 206IR-101 Support Installation & IR-605 Support Plate		
		Notes: 1. Only structural aspects of the above data are approved herein. This approval is for engineering design data only and is not an installation approval. It indicates the data listed above demonstrates compliance with the regulations specified by paragraph and subparagraph listed below as 'APPLICABLE REQUIREMENTS'. (Compliance to additional regulations not listed here may be required). This form does not constitute FAA approval of all engineering data necessary for substantiation of compliance to necessary requirements for the entire alteration/repair. 2. This approval is valid for Cessna Aircraft Company Model 206 S/N T20608983		
8. PURPOSE OF DATA Submittal of data in support of FAA Major Alteration				
9. APPLICABLE REQUIREMENTS (List specific sections) 14 CFR 23.301, 23.303, 23.305(a), 23.307(a), 23.337(a), 23.601(a), 23.603, 23.605(a), 23.607(a), 23.609(a)(2), 23.611, 23.613, & 23.625(a).				
10. CERTIFICATION -Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183 of the Federal Aviation Regulations, data listed above and on attached sheets numbered <u>2</u> have been examined in accordance with established procedures and found to comply with applicable requirements of the Federal Aviation Regulations.				
I Therefore Recommend approval of these data <input checked="" type="checkbox"/> Approve these data				
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)		12. DESIGNATION NUMBERS(S)	13. CLASSIFICATION(S)	
 Joe Musco		DERT-605388-NM	Structures	



Applicable Requirement Amendment Levels:

FAR	Title	Amdt.
23.301	Loads	23-42
23.303	Factor of safety	-
23.305(a)	Strength and deformation	-
23.307(a)	Proof of structure	-
23.337(a)	Limit maneuvering load factor	-
23.601(a)	Design	-
23.603	Materials	-
23.605(a)	Fabrication methods	-
23.607(a)	Fasteners	-
23.609(a)(2)	Protection of structure	-
23.611	Inspection provisions	23-7
23.613	Material strength properties and design values	-
23.625(a)	Fitting factors	-



FLAMMABILITY REPORT, TEST, AND TEST RESULTS

**PARAVION TECHNOLOGY, INC.
2001 AIRWAY AVENUE
FORT COLLINS, CO 80524**

LOG OF REVISIONS

Revision	Date	Description	By
0	3/15/2013	Original	Douglas White
1	4/8/2013	Added test results	Douglas White

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1. PURPOSE

This report defines the plan for the testing of a carbon fiber composite material, which will be used in the center console of a Cessna 206 aircraft. Upon completion of testing, this document will be revised to include test results and supporting data.

2. REGULATORY REQUIREMENTS

The flammability requirement for a composite console installed in compartment interiors, as defined in the certification basis for the Cessna 206, is as follows (Ref. TCDS 3A13):

14 CFR 23.853, Amendment 23-62:

Compartment interiors

For each compartment to be used by the crew or passengers:

(a) The materials must be at least flame-resistant

To meet this requirement, testing will be conducted in accordance with Part 25 Appendix F paragraph (a)(1)(ii), 12-second vertical.

3. TEST ARTICLE CONFIGURATION

The components to be qualified are manufactured from a carbon fiber resin composite, configured as follows:

- 1x1 Plain Weave Carbon (visible layer) – from CST
- 2x2 Twill 3K Carbon (inner layers) – from US composites
- LAM-135-FR Resin – from Pro-set
- LAM-229 Hardener – from Pro-set

In accordance with Appendix F of Part 25, the test articles will be cut from stock material, without finished or protected edges, representative of the actual cross-section and thickness of the part as installed in the aircraft.

The overall size of the specimen will be 3 inches wide and 13 inches long, exposing an area 2 inches wide and 11 inches long to the flame. Three specimens will be tested.

4. TEST FACILITY AND EQUIPMENT

The test facility is located at Centennial Aircraft Interiors Annex, 12559 E. Broncos Pkwy, Centennial, CO 80112. The following equipment will be used to perform the

flammability tests described in this document. The Vertical Flammability Test Cabinet meets the requirements of 14 CFR Part 25, Appendix F, Part I, Section (b)(3) and DOT/FAA/AR-00/12, Aircraft Materials Fire Test Handbook.

Equipment Nomenclature	Model Number	Manufacturer
Conditioning Chamber	Stabil-Therm Laboratory Oven	Blue M Electric Co. Blue Island, IL
Vertical Flammability Test Cabinet	7633A	United States Testing Co., Inc. Hoboken, NJ

5. TEST ARTICLE CONDITIONING

The test articles will be conditioned to $70^{\circ} \pm 5^{\circ}\text{F}$ and at $50\% \pm 5\%$ relative humidity until moisture equilibrium is reached or for 24 hours. Each specimen must remain in the conditioning environment until it is subjected to the flame.

6. CONFORMITY INSPECTIONS

Company conformity of the test articles will be conducted by Paravion Technology and documented on FAA Form 8130-9, Statement of Conformity. The test set-up will be verified by the witnessing Flammability DER to be in accordance with this test plan. The witnessing DER will coordinate with Paravion Technology, Inc, if design data changes are necessary, prior to DER approval.

7. VERTICAL TEST SET UP CONFIGURATION

The vertical test will be configured and conformed as follows, in accordance with 14 CFR Part 25, Appendix F, Part I, Section (b)(4):

- A Bunsen burner with a nominal 3/8 inch I.D. tube will be used for the test.
- Prior to testing, ignite the burner and set the flame height to 1½ inches.
- Using a calibrated thermocouple pyrometer, verify that the minimum flame temperature in the center of the flame is 1550° F . Record the flame temperature on the Test Data Sheet in Appendix B, and extinguish the flame.
- Set the automatic timer on the Flame Control Module to 12.0 seconds.
- Verify that the conditioning chamber has maintained the test articles at $70^{\circ} \pm 5^{\circ}\text{F}$ and $50\% \pm 5\%$ relative humidity for a minimum of 24 hours.
- Remove one test article from the conditioning chamber.

- Position the specimen in the support frame of the vertical test stand so that the edge being tested is centered $\frac{1}{4}$ -inch above the top of the burner and the flame is applied to the center of the specimen. See Fig 1.
- Close the test cabinet door.



Fig 1. Test Specimen in Support Frame and Placement in Vertical Test Stand

8. TEST CONDUCT PROCEDURES

The vertical test will be conducted as follows, in accordance with 14 CFR Part 25, Appendix F, Part I, Section (b)(4). A minimum of three samples must be tested, and the results averaged.

- Verify the test article has been positioned in the chamber in accordance with Section 7.0.
- Activate the automatic flame timer switch.
- Verify that the burner ignites and the flame is applied to the center of the edge of the specimen.
- Verify that the automatic flame timer extinguishes the flame after 12 seconds.
- Observe the behavior of the specimen after the burner flame is extinguished. Continue timing as long as the specimen continues to flame. Note any drippings from the specimen and the flame time of the drippings.
- Record the flame time, burn length (to the nearest 0.1 inch) and flame time of drippings on the test data sheet.

9. PASS/FAIL CRITERIA

The material is considered to pass this test if all of the following criteria are met:

- The average flame time of the specimen after removal of the flame source may not exceed 15 seconds.
- The average burn length may not exceed 8 inches.
- Drippings from the test specimen may not continue to flame for more than an average of 5 seconds after falling.

10. TEST WITNESSING AND DATA APPROVAL

The selected Flammability DER will witness the tests and approve the test results. A copy of the approved test report and 8110-3 will be forwarded to Paravion Technology, Inc.

11. DESIGNATED PERSONNEL

The following is a list of designated personnel to be involved in this project:

Title	Name
DER Flammability	Bob Lazaroff DERT-660022-NM

12. REFERENCES

The following documents form a part of this document to the extent specified herein:

Document	Description
14 CFR Part 23	Airworthiness Standards, Normal Category Aircraft
14 CFR Part 25	Airworthiness Standards, Transport Category Aircraft
14 CFR Part 25, Appendix F, Part I	Test Criteria and Procedures for Showing Compliance with Flammability Requirements
DOT/FAA/AR-00/12	Aircraft Materials Fire Test Handbook
FAA Order 8110-113	Approval of Flammability Test Data in Support of Major Repairs or Major Alterations
Type Certificate Data Sheet 3A13	Type Certificate for Cessna Model 206

13. APPENDIX A: TEST RESULTS REPORT FROM MR. LAZAROFF

TEST SUMMARY

Testing was conducted on April 8, 2013 at the Centennial Aircraft Interiors Annex, 12559 E Broncos Pkwy, Centennial, CO 80112 by DER Bob Lazaroff, who was also delegated by the Denver ACO to witness the tests on their behalf. The test set-up was verified by the witnessing DER.

Three test articles were placed into the conditioning chamber, at 70°F and 50% humidity, at 1030 on April 6, 2013. The test articles remained in the conditioning chamber until they were removed one at a time for testing.

The test articles were tested vertically in accordance with Sections 7 and 8 of this test plan. All three test articles self-extinguished after flame removal, with an average burn length of 1.50 inches, and with no flaming drips, thus successfully meeting the requirements of 14 CFR 23.853(a) as stated in Section 2 of this test plan.

Cessna 206
Console

FLAMMABILITY TEST REPORT
DATA SHEET

C206-CF Console

Test Article ID/PN	Material	Representative of: (list part number and description)	Regulatory Requirement and Amendment Level	Test Criteria per Part 25 Appendix F, Part I (mark "X" in appropriate block)			
				(a)(1)(i) 60 second Vertical	(a)(1)(ii) 12 second Vertical	(a)(1)(iv) 15 second Horizontal 2.5 in/min	(a)(1)(v) 15 second Horizontal 4.0 in/min
1, 2, 3	1x1 Plain Weave Carbon 2x2 Twill 3K Carbon LAM-135-FR Resin LAM-229 Hardener	Console	23.853(a) Amdt 23-62 *		X		

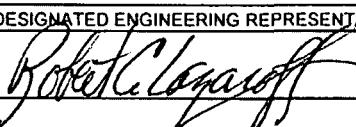
Test Sample	Test Results per Part 25 Appendix F, Part I (enter results in appropriate block)							
	(a)(1)(i) 60 second Vertical			(a)(1)(ii) 12 second Vertical			(a)(1)(iv) 15 second Horizontal	(a)(1)(v) 15 second Horizontal
	Burn Length (< 6 in)	After-Flame (< 15 sec)	Drip Flame (< 3 sec)	Burn Length (< 8 in)	After-Flame (< 15 sec)	Drip Flame (< 5 sec)	Burn Rate (< 2.5 in/min)	Burn Rate (< 4.0 in/min)
1				1.75 in	10.8 sec	No drips		
2				1.75 in	11.1 sec	No drips		
3				1.0 in	2.9 sec	No drips		
Average				1.50 in	8.3 sec	No drips		

* Tested to Part 25 requirements

Results: PASS FAIL

DER: Robert Clagett DEET-660022-NM

14. APPENDIX B: FAA FORM 8110-3

U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION STATEMENT OF COMPLIANCE WITH AIRWORTHINESS STANDARDS					1. DATE 04/10/2013
AIRCRAFT OR AIRCRAFT COMPONENT IDENTIFICATION					
2. MAKE Cessna	3. MODEL NO. T206	4. TYPE (Aircraft, Engine, Propeller, etc.) Aircraft	5. NAME OF APPLICANT Paravion Technology, Inc.		
LIST OF DATA					
6. IDENTIFICATION	7. TITLE				
Report No. C206-CF Console Revision 1, 04/08/2013	Flammability Report, Test and Test Results				
<p>Notes:</p> <p>1. All engineering aspects of the above listed data are approved herein. This approval is for engineering data only. It indicates the data listed above demonstrates compliance only with the regulation specified by paragraphs and subparagraph listed below as "Applicable Requirements".</p> <p>2. This approval is for flammability aspects of the proposed installation only. Additional approvals may be required for the substantiation of compliance to necessary requirements for the entire type design change.</p> <p>3. Delegation to approve test plan and witness tests by Satish Lall, Denver ACO, per Special Authorization, 4/1/13.</p> <p>4. This approval is valid for the installation of the Hand Control Connector Housing on Cessna Model T206 S/N T20608983 only</p>					
8. PURPOSE OF DATA In support of a major alteration to Cessna Model 206 S/N T20608983					
9. APPLICABLE REQUIREMENTS (List specific sections) 14 CFR 23.853(a), Amendment 23-62					
10. CERTIFICATION - Under authority vested by direction of the Administrator and in accordance with conditions and limitations of appointment under 14 CFR Part 183, data listed above and on attached sheets numbered <u>N/A</u> have been examined in accordance with established procedures and found to comply with applicable requirements of the Airworthiness Standards listed.					
<input type="checkbox"/> Recommend approval of these data I (We) Therefore <input checked="" type="checkbox"/> Approve these data					
11. SIGNATURE(S) OF DESIGNATED ENGINEERING REPRESENTATIVE(S)		12. DESIGNATION NUMBERS(S)		13. CLASSIFICATION(S)	
Robert C. Lazaroff 		DERT-660022-NM		Structures	





The Composites Store, Inc.

COPY

PRODUCT CONFORMANCE CERTIFICATION

Customer No.: R17364

Ship Date: 03/11/2013

Sold To: PARAVION TECHNOLOGY, INC. **Ship To:** SAME

Purchase Order No. 43778

Invoice No.: 130964

Item No.	Description	Quantity
CF141	3.5 oz. Carbon Fabric, 42" wide, Plain Weave	9 ft.
FDI Style Number: 824		
Lot Number: 22032	Yarn Type: T300 1K_309 NT	
Roll Number: 6A	Date of Mfg: 08/2012	
Weave: PW, Count: 24.1 x23.2 inches;	Width: 42 ¾ “;	Weight: 126.0 g/m ²
Thickness: 0.008 inches		

We hereby certify that the material listed conforms to applicable commercial specifications, or government specifications as shown below.

SPECIFICATION: Commercial Grade

Gail Gewain, President

Website: www.cstsales.com

P.O. Box 622, Tehachapi, California 93581-0622 • Phone: 661-822-4162 • Fax 661-822-4121



U.S. COMPOSITES, INC.
561/588-1001
6670 WHITE DRIVE
WEST PALM BEACH FL 33407

* * * Invoice 256020 * * *

Bill To : Ship To : Customer No. 270051
Paravion Technology, Inc.
Valerie McAlpine
2001 Airway Ave.
FORT COLLINS 80524

DATE	ORDER NO.	SLS.NO.	ORDER DATE	PURCHASE ORDER	SPECIAL INSTRUCTIONS
03/28/13			03/28/13		

Quantity U/M	Description Stock Number	Code	Price	Extension
1.00 ea	CERT FEE FOR PREVIOUS SHIPMENT INVOICE # 255095	0004	\$10.00	\$10.00
	LOT # 1233808 ROLL # 0003802464			

Payment/Terms :	SUBTOTAL :	\$10.00
prepaid credit card	TAX :	
	FREIGHT :	
	INVOICE TOTAL DUE =====>	\$10.00

From the Quality Control Laboratories of:



U S COMPOSITES INC
6670 WHITE DRIVE
WEST PALM BEACH FL 33407

Date: 01/29/13 9:35:25
Page: 1

Attention:

Style: 94933
Width: 50.0
Code/Part# Contract: KB00219873 10

Cust P.O.: 403389
Weave: 2X2 TWILL

Lot Number: 01233808/00010

Yarn Type Warp: TR30S 3K

Fill: TR30S 3K

Warp Lot: 1233303A
Fill Lot: 1223305A 1233303A 1253304A

Lot#	Cut	Roll #	LBS	Yards	Meas.		Warp Yarn	Fill Yarn	Dry Weight G/M2	Ext. Weight G/M2	Thickness Inch	Date of MFG.
					Net Wgt	Width Inch						
					Inch	/Inch						
1233808	009	0003802303	55	100	50.1	13.1	13.5	211.0	208.6	.0090	012513	
1233808	011	0003802308	52	100	50.1	13.1	13.5	210.4	208.0	.0095	012513	
1233808	013	0003802463	52	100	50.1	13.1	13.5	208.6	206.2	.0093	012513	
1233808	012	0003802464	52	100	50.1	13.1	13.5	210.4	208.0	.0095	012513	
1233808	017	0003803687	53	100	50.1	13.1	13.5	211.4	209.0	.0097	012813	
1233808	016	0003803688	50	100	50.1	13.1	13.5	211.2	208.8	.0092	012813	
Total:			314	600								

We Certify that this material meets the requisites of

BGF DATA SHEET

Prepared and Verified by

Robert Gatti
Authorized Quality Representative

From the Quality Control Laboratories of:



U S COMPOSITES INC
6670 WHITE DRIVE
WEST PALM BEACH FL 33407

Date: 01/29/13 9:35:25
Page: 2
Style: 94933
Width: 50.0
Code/Part#
Contract: KB00219873 10

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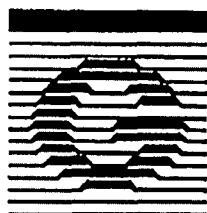
Cust P.O.: 403389
Weave: 2X2 TWILL

Lot Number: 01233808/00010

Yarn Type Warp: TR30S 3K

Fill: TR30S 3K

Lot #	Cut	Roll #	Shipment Defect Summary		Yard Length	Defect
			Yards	Distortion		
1233808	009	0003802303	100	.1	6	DISTORTION
1233808	011	0003802308	100	.0		No Defects Recorded
1233808	013	0003802463	100	.1		No Defects Recorded
1233808	012	0003802464	100	.0		No Defects Recorded
1233808	017	0003803687	100	.2		No Defects Recorded
1233808	016	0003803688	100	.0		No Defects Recorded

**GRAFIL INC.**

Certificate of Conformity

5900 88th Street
Sacramento, CA 95828 USA

Order No. 29952

Phone No. 916/386-1733 or 800/365-5533
Fax No. 916/379-2183

Customer No. BGF01

Deliver To: BGF Industries Inc.
Cheraw Speciality Weaving
90 Huger Street
90 Huger Street
Cheraw, SC 29520

Certificate of Conformity: 23193

GI Reference: 29952

Certificate Date: 10/25/2012

Fiber Type: TR30S 3K 1.2%
PYROFILSize: 1.2%
S

Quantity in lbs: 3,531.52

Customer Purchase Order		Item #	Specification			Salesperson	Customer Part #	
03005814-100		TR-30S-3SL	CF-202 Ver.1			Wayne Schaefer		
Batch No.	Date of Manufacture	Quantity (lbs)	Strength (ksi)	Modulus (msi)	Yield (yd/lb)	Fiber Density (lb/in ³)	Size Content (% by Mass)	Elongation (%)
1223305A	02/2012	3,531.52	604.8	33.5	2469.0	0.06429	1.2	1.80
Shelf Life for Pyrofil: 3 years from Date of Manufacture								
<p>Certified that the supplies/services detailed herein have been inspected and tested in accordance with the conditions and requirements of the contract or conform in all respects to the specification(s), drawings relevant thereto.</p> <p>Signed: _____</p> <p><i>R. Argent</i></p> <p>For and on behalf of Grafil Inc.</p>								



Certificate of Conformity

6900 88th Street
Sacramento, CA 95828 USA

Phone No. 916/386-1733 or 800/365-5533
Fax No. 916/379-2183

Order No. 30586

Customer No. BGF01

Deliver To: BGF Industries Inc.
Cheraw Specialty Weaving
90 Huger Street
90 Huger Street
Cheraw, SC 29520
USA

Certificate of Conformity: 23444

GI Reference: 30586

Certificate Date: 01/16/2013

Fiber Type: TR30S 3K 1.2%S

PYROFIL

Size: 1.2%S

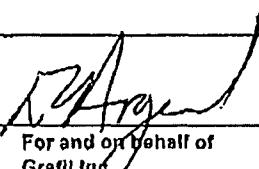
Quantity In lbs: 3,491.84

Customer Purchase Order		Item #	Specification			Salesperson	Customer Part #	
03006103-20		TR-30S-3LSL	CF-202 Ver.1			Wayne Schaefer		
Batch No.	Date of Manufacture	Quantity (lbs)	Strength (ksi)	Modulus (mci)	Yield (yds/lb)	Fiber Density (lb/in³)	Size Content (% by Mass)	Elongation (%)
1233303A	03/2012	357.12	604.0	33.8	2469.0	0.06429	1.2	1.80
1263304A	05/2012	3,134.72	603.0	33.4	2460.0	0.06419	1.2	1.80
	Cert total:	3,491.84						

Shelf Life for Pyrofil: 3 years from Date of Manufacture

Certified that the supplies/services detailed herein have been inspected and tested in accordance with the conditions and requirements of the contract or conform in all respects to the specification(s), drawings relevant thereto.

Signed: _____



For and on behalf of
Grafil Inc.

2025 RELEASE UNDER E.O. 14176

Monday, March 11, 2013

PRO-SET

To Whom it May Concern:

This is to certify that the PRO-SET® product(s) that you recently purchased were manufactured in accordance with our standard quality control procedures. In addition, a representative sample from each batch was tested for conformity to our internal specifications and a portion of that sample will be retained for 18 months from the production date.

RESULTS OF ANALYSIS

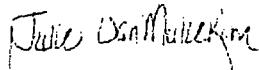
<u>Product:</u>	<u>Batch Number:</u>	<u>Date Produced</u>	<u>Tested With:</u>
LAM-135-FR	1363018A	01-18-2013	LAM-224
<u>DSC 822e</u>			<u>Limits</u>
TPeak1=85.69(°C)			85.38-86.73(°C)
TPeak2=0(°C)			-(°C)
dH=489.44(J/g)			434.82-505.34(J/g)
tg=90.43(°C)			89.77-93.4(°C)
<u>Viscosity</u>			<u>Limits</u>
3972cPs			3662-4027cPs

STATEMENT OF SHELF LIFE

Minimum 3 years from date of manufacture when stored in original sealed container.

Sincerely,

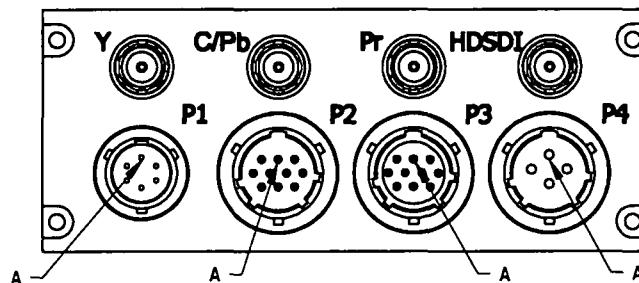
PRO-SET, INC.



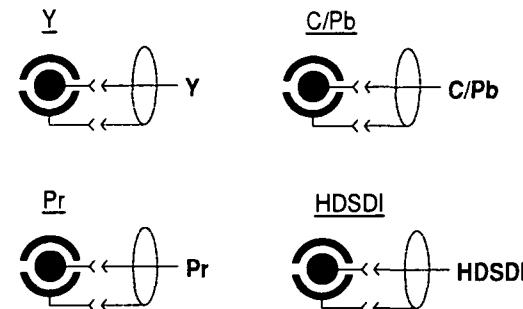
51. *Scutellaria* *barbata* L.

Scutellaria barbata L. *Scutellaria* *barbata* L. *Scutellaria* *barbata* L.

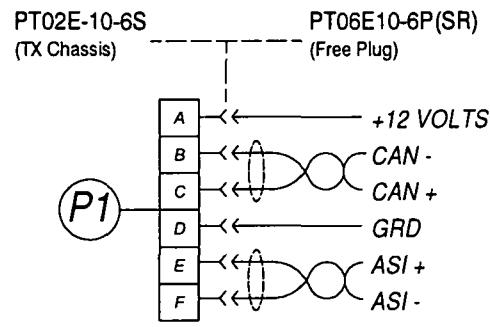
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NOTES:
VOLTAGE RANGE: 11 - 30 VOLTS
POWER CONSUMPTION: 72 VA
TROLL INTERFACE: CONNECTOR P2 PINS H, J & K
FOR REFERENCE USE ONLY



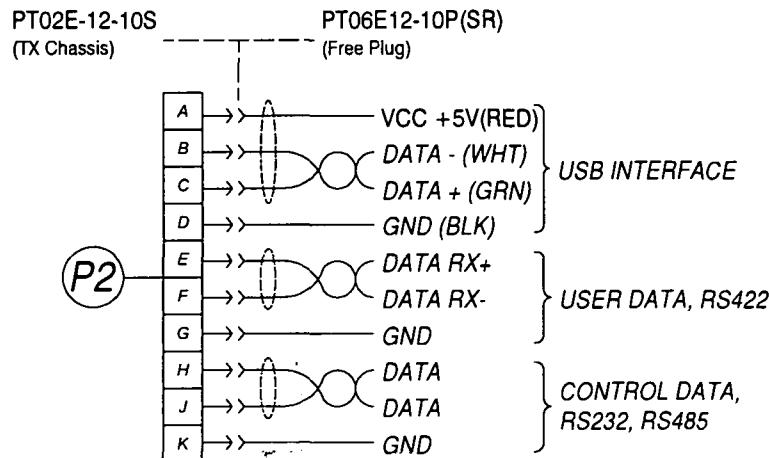
REVISION HISTORY			
REV	DESCRIPTION	DATE	APPROVED
A	INITIAL RELEASE	3/16/2011	O REYES



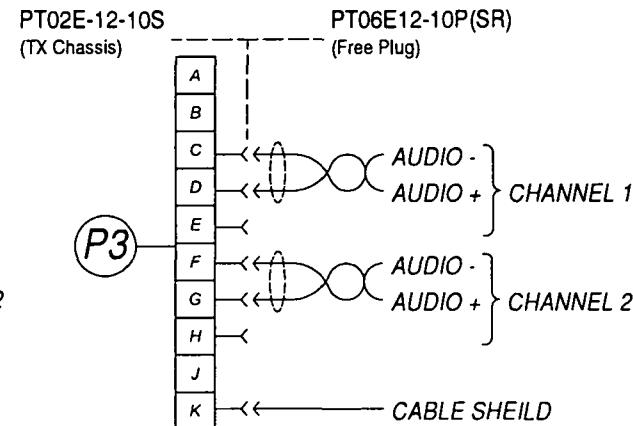
HAND HELD CONTROLLER



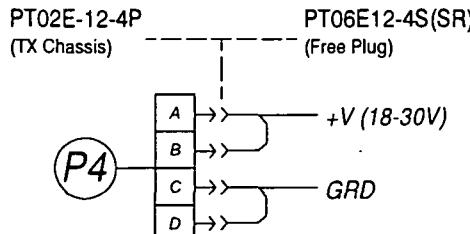
DATA INTERFACE



AUDIO INTERFACE



POWER INTERCONNECT



TOLERANCE INFO:
ALL DIMENSIONS ARE IN INCHES.
DECIMAL .XX ±.001, .XXX ±.0025, ANGULAR ±.3°
HOLE SIZE ±.003, .002, CONCENTRICITY ±.001,
FLATNESS ±.0025, SURFACE FINISH 63 OR BETTER

CONFIDENTIAL INFO:
THIS DOCUMENT AND INFORMATION IT CONTAINS IS
CONFIDENTIAL AND THE SOLE PROPERTY OF JANTEQ
INC. AND MAY NOT BE REPRODUCED OR DISTRIBUTED
WITHOUT THE WRITTEN CONSENT OF JANTEQ INC.

PART NO: 1011139
REV: A
DESIGNER: J PORTER
ENGINEER: O REYES
SHEET: 1 OF 1
DATE: 3/16/2011

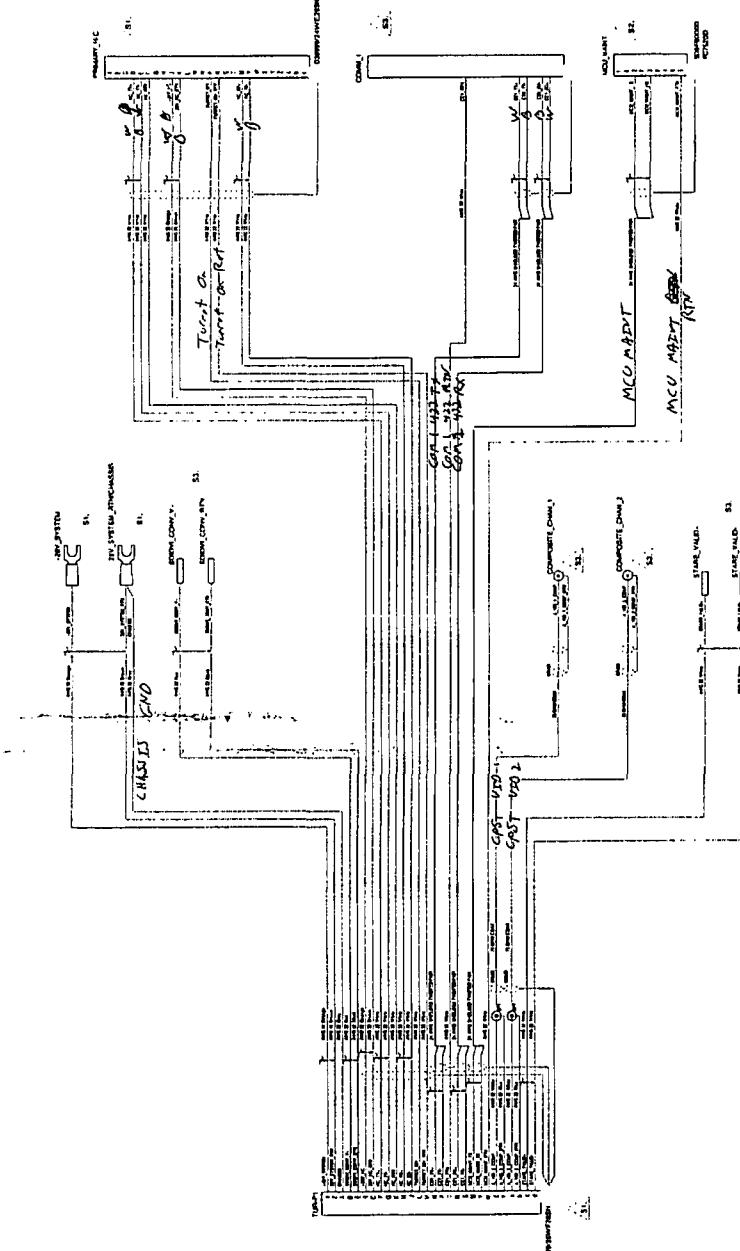
AVIATION Tx WIRING
INTERFACE

JANTEQ
CORPORATION
9272 JERONIMO RD.
IRVINE, CA. 92618
PH: (949) 215-2603
FAX: (949) 215-2604

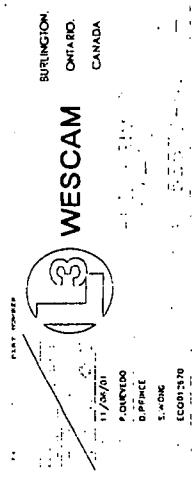


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Use is granted to the United States Government
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only, and is subject to the terms and conditions
of the contract under which it was furnished.

Document Number: 55571-01
Date: 2/24/01
Rev: A
Description: REVIT 045
Title: APP C
Date: 05/15/01



- NOTES:**
- S1: SPECIFIED CONNECTOR IS REQUIRED.
 - S2: RECOMMENDED CONNECTOR WOULD BE USEFUL FOR WESCAM PROBLE SHOOTING.
HOWEVER IT IS NOT REQUIRED.
 - S3: OPTIONAL REQUIREMENT DEPENDANT ON INDIVIDUAL SYSTEM CONFIGURATION
AND OPTIONS. SPECIFIED CONNECTOR IS REQUIRED IF CONFIGURATION IS DESIRED.
 - S4: ENSURE GOOD ELECTRICAL CONDUCTIVITY BETWEEN SHIELDS AND BACKSHELL.
 - REFERENCE DOCUMENT: #4370 GDO FOR MD-10 DIGITAL TURNET.
 - REFERENCE ONLY. GDO TAKES PRECEDENCE.
 - MARSHAL WIRE GUARD IS RECOMMENDED.



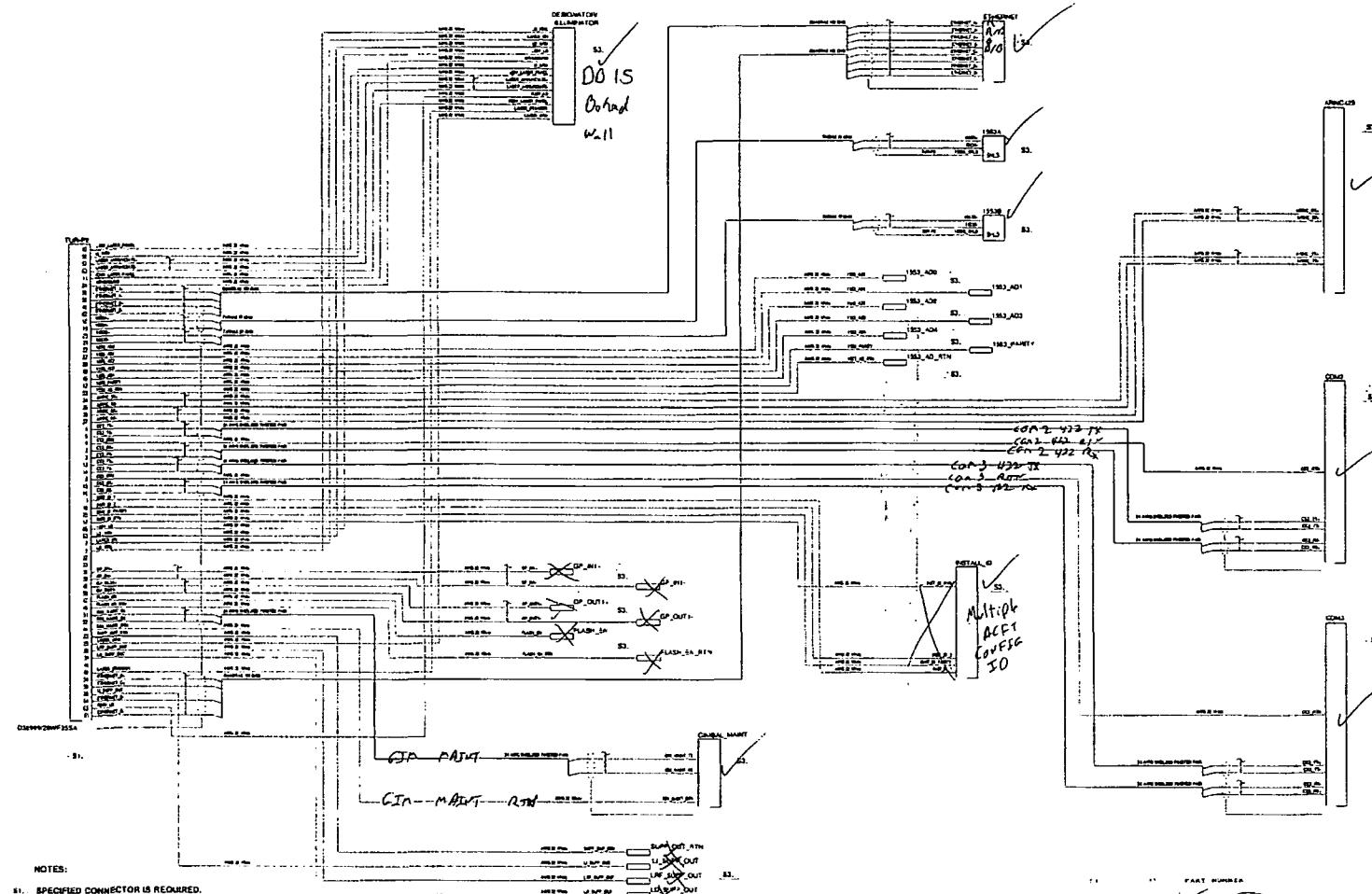
11/24/01
P. COLEMAN
D. PARFET
S. LANGE
EC01@2GO.CA
WWW.WESCAM.COM
S5571-01S
1 OF 1

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15124 52573-01

1000 202.

REVISIONS
DESCRIPTION DATE APPROVED



L3 WESCAM

BURLINGTON,
ONTARIO,
CANADA

P. OUTWOOD
D. PRINCE
S. ADIGIC
EC0013670
52573-01-035

11/06/01
HIS

SHEET 1 OF 1

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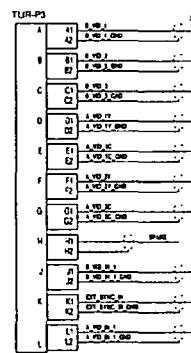
COMPUTER GENERATED
DO NOT REVISE MANUALLY
DO NOT SCALE DRAWING

SAC24 53574-01

ZONE - REV.

REVISIONS
DESCRIPTION

DATE APP'D



NOTES:

S1. SPECIFIED CONNECTOR IS REQUIRED.

S2. RECOMMENDED; CONNECTOR WOULD BE USEFUL FOR WESCAM TROUBLE SHOOTING,
HOWEVER IT IS NOT REQUIRED.

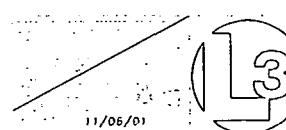
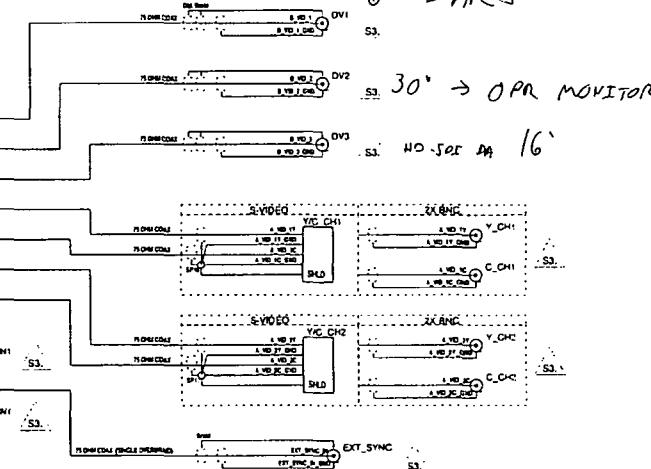
S3. OPTIONAL; REQUIREMENT DEPENDANT ON INDIVIDUAL SYSTEM CONFIGURATION
AND OPTIONS. SPECIFIED CONNECTOR IS REQUIRED IF CONFIGURATION IS DESIRED.

ENSURE GOOD ELECTRICAL CONDUCTIVITY BETWEEN SHIELDS AND BACKSHELL.

MINIMUM WIRE GAUGE IS IDENTIFIED.

REFERENCE DOCUMENT: 64270 ICD FOR MX-10 DIGITAL TURRET.

REFERENCE ONLY, ICD TAKES PRECEDENCE.



WESCAM

BURLINGTON,
ONTARIO,
CANADA

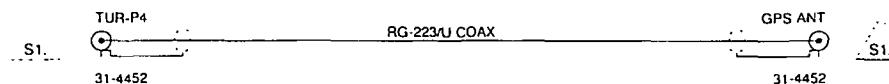
11/06/01
P.QUEVEDO
D.PRINCE
S.WONG
EC0013670
535741-E3S NTS

SHEET 1 OF 1

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ZONE	REV.	REVISIONS	DESCRIPTION	DATE	APP'D
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NOTES:

S1. SPECIFIED CONNECTOR IS REQUIRED.

S2. RECOMMENDED; CONNECTOR WOULD BE USEFUL FOR WESCAM TROUBLE SHOOTING,
HOWEVER IT IS NOT REQUIRED.

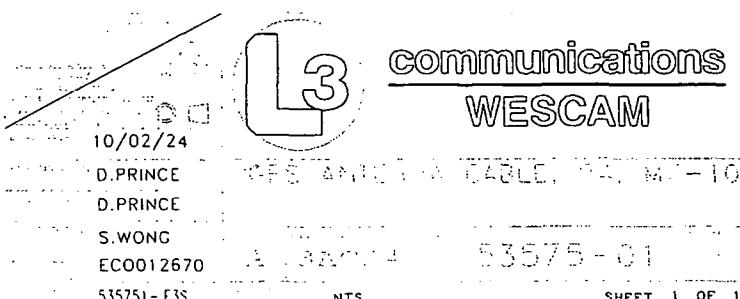
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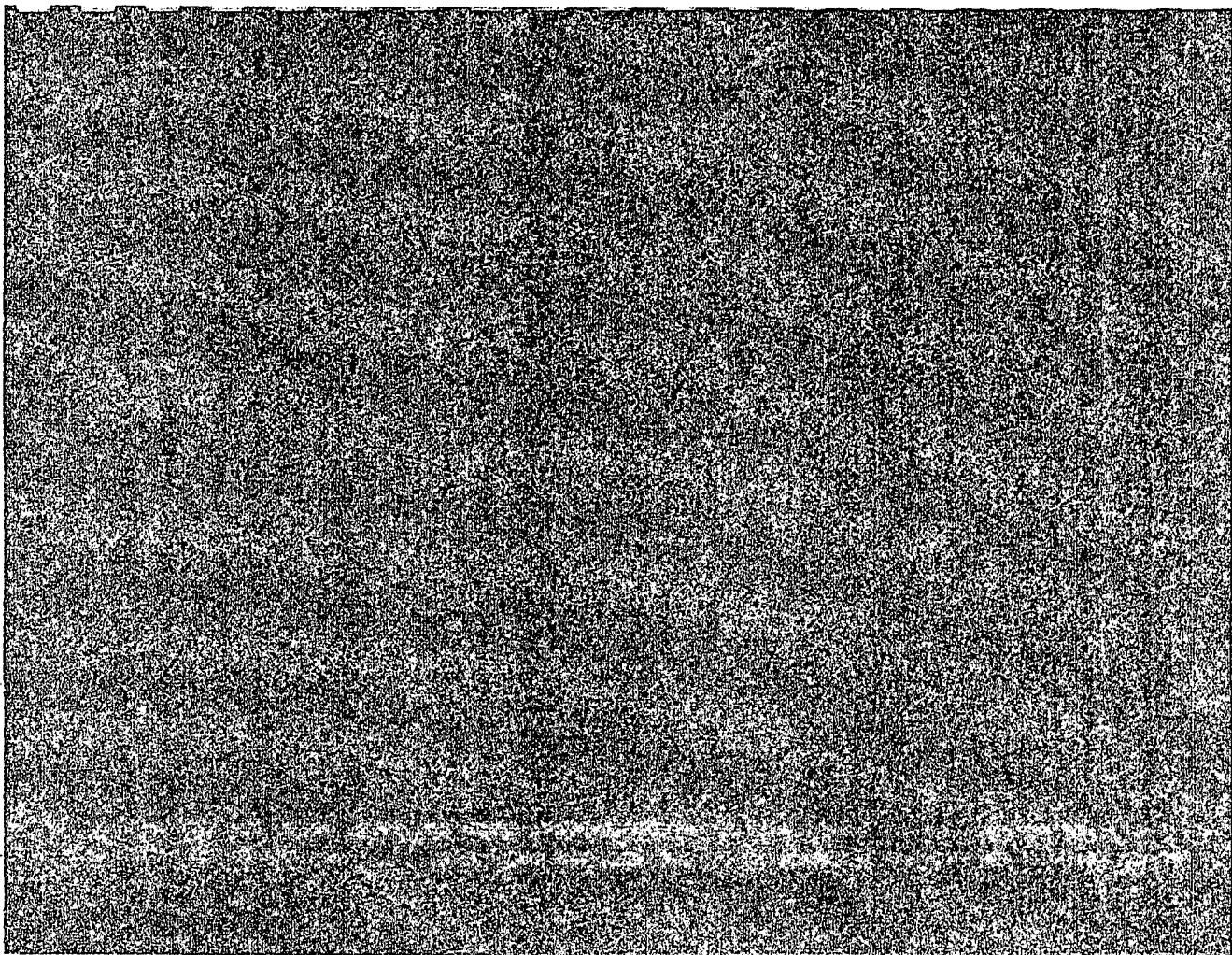
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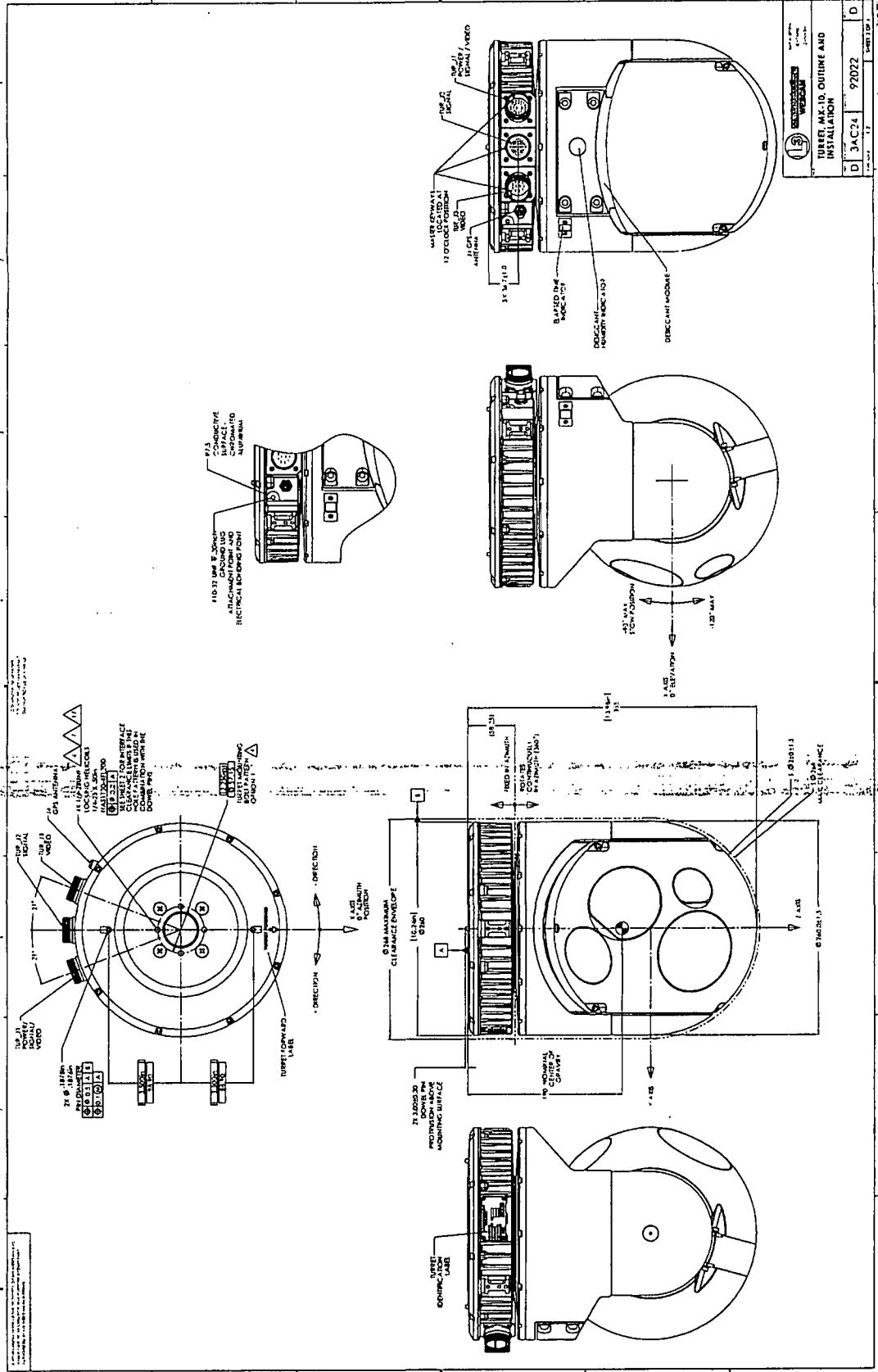
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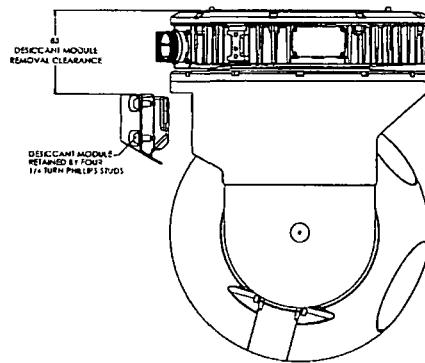
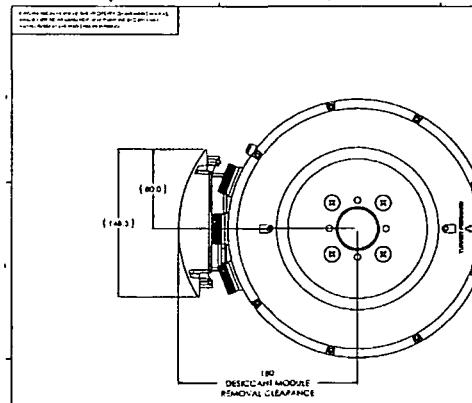
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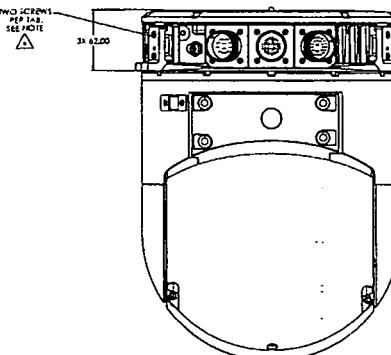
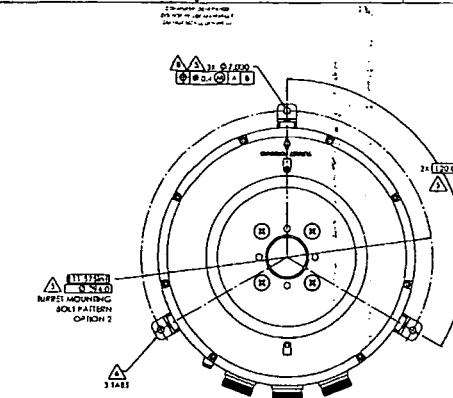




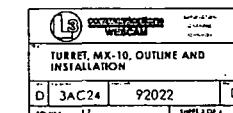
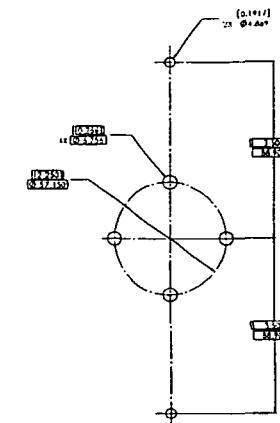


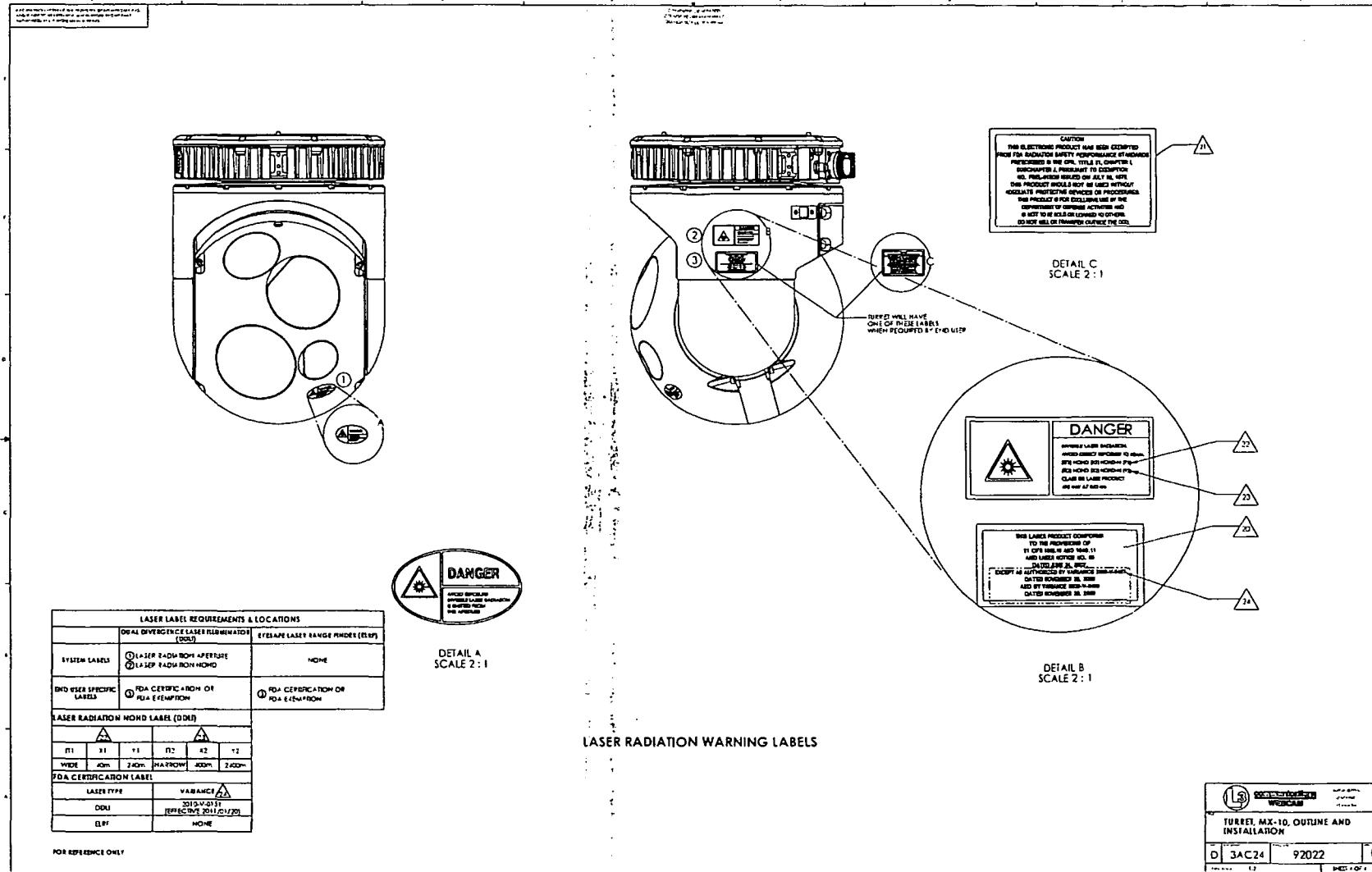


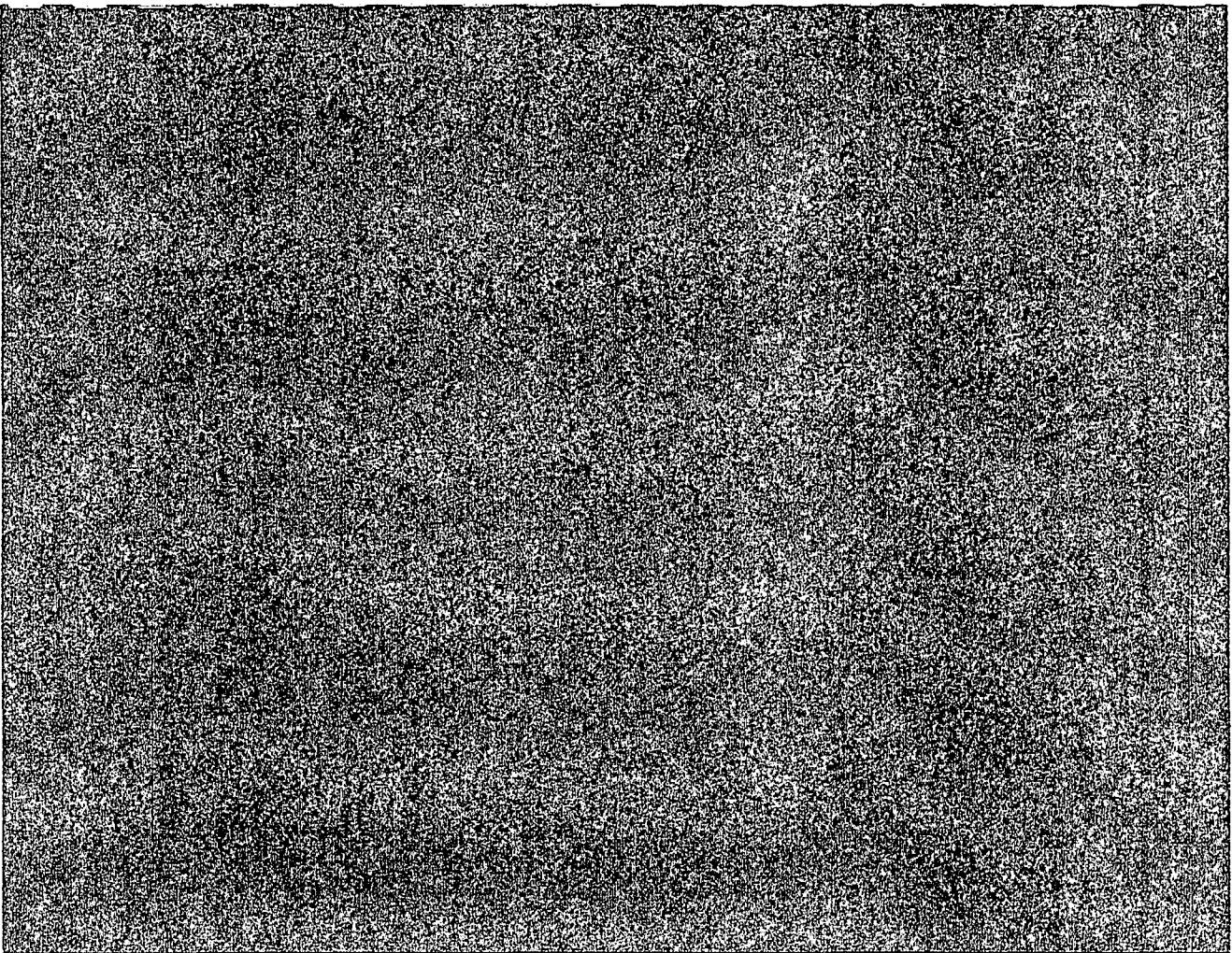
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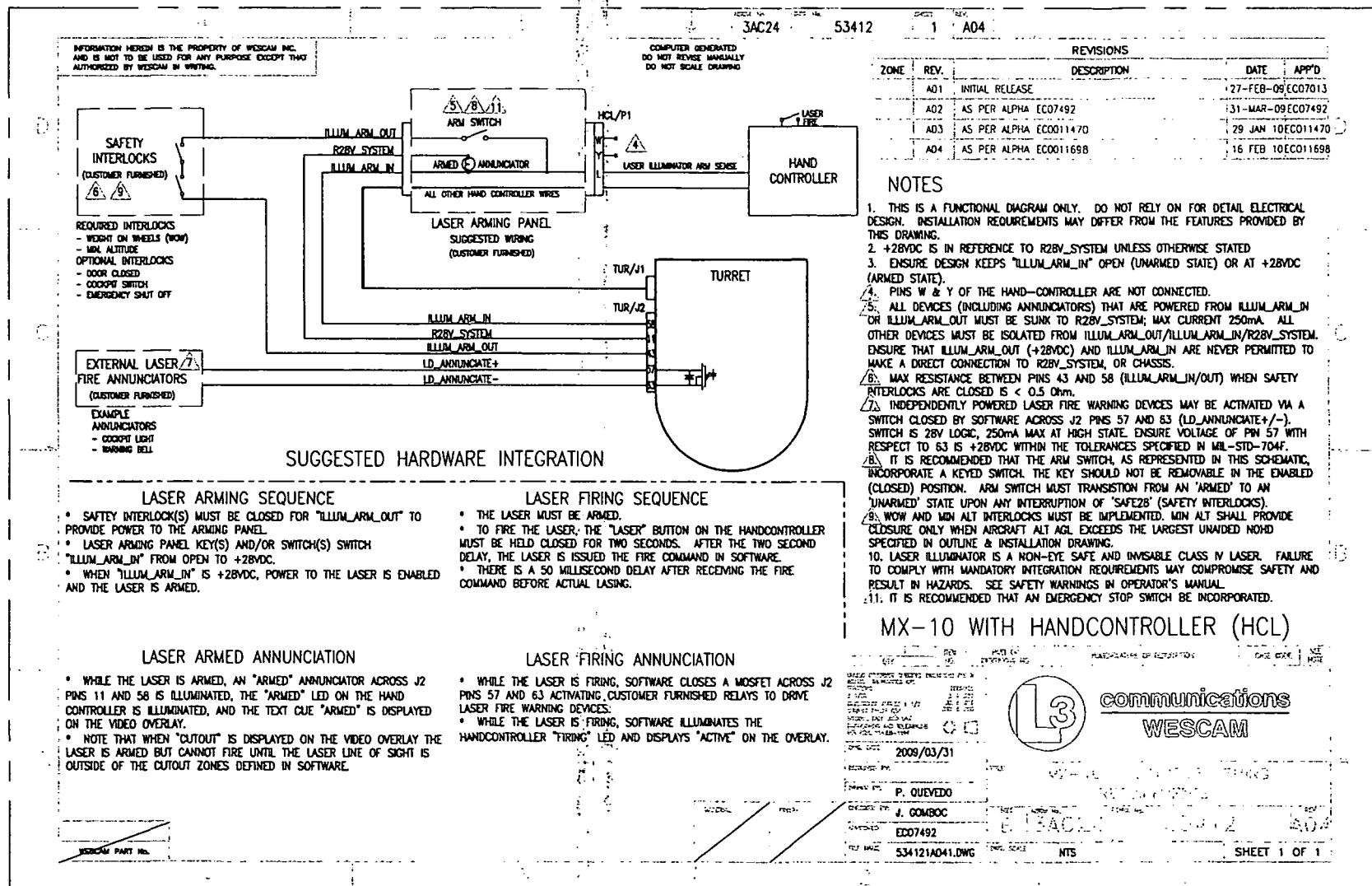
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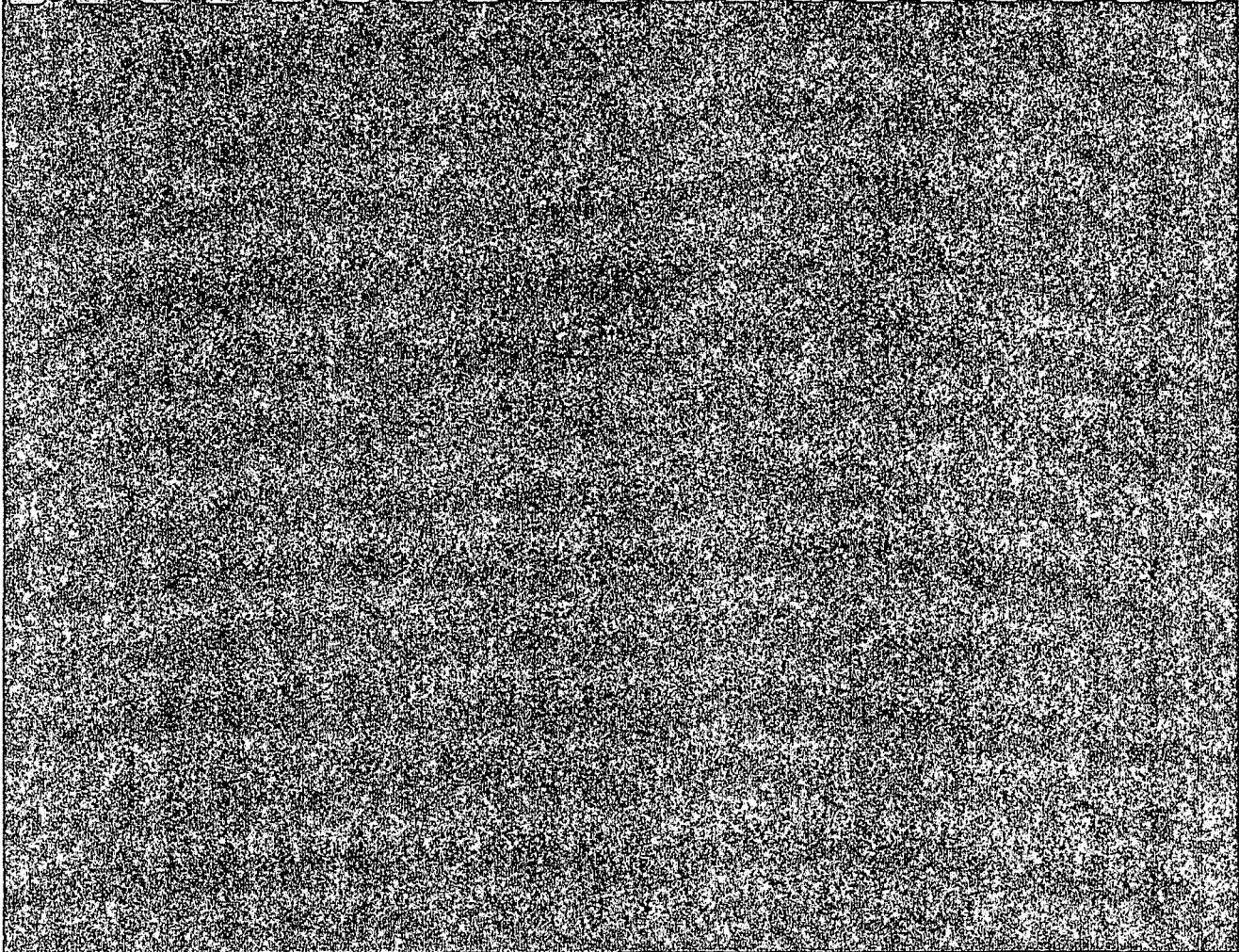






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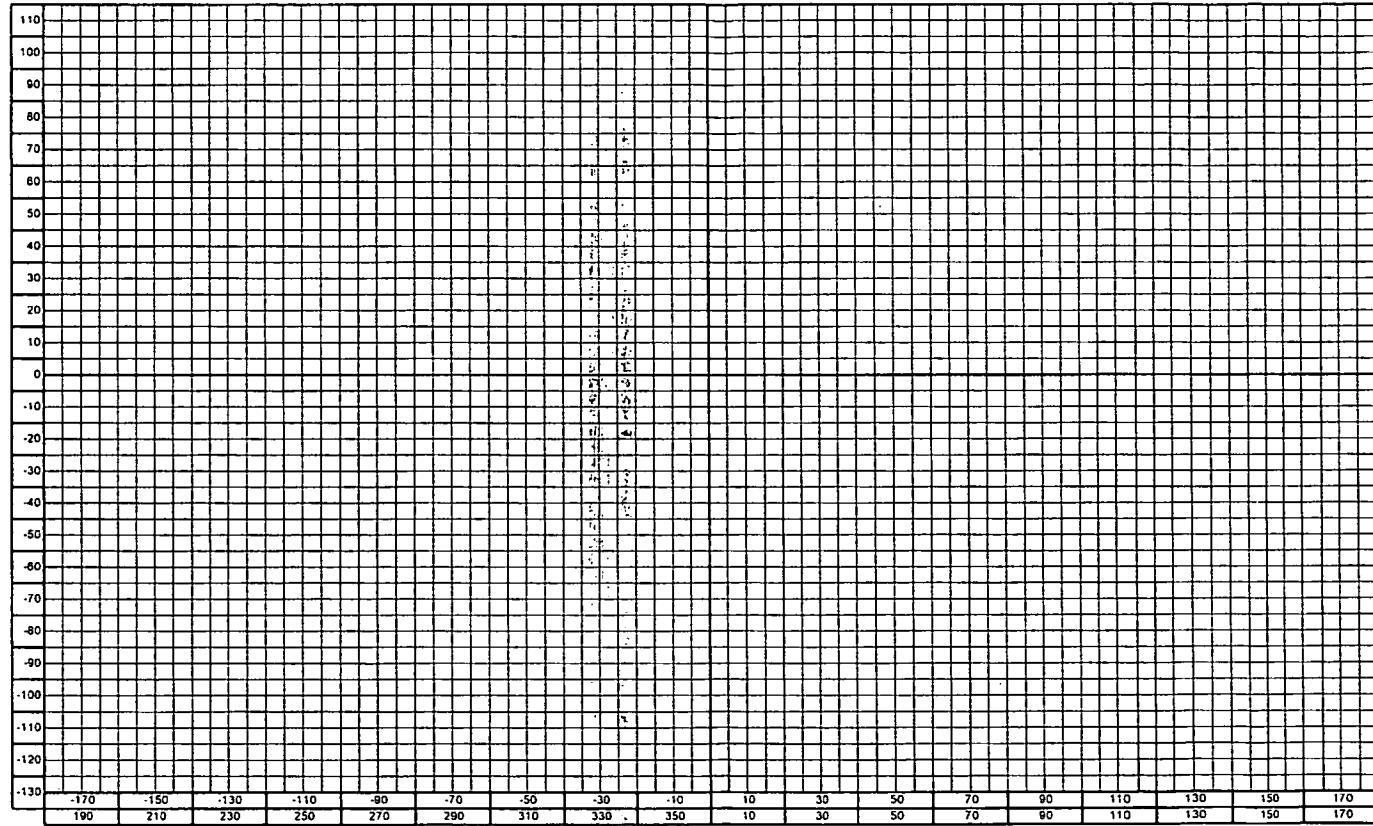
LASER CUTOUT DEFINITION FORM

FM1287
Rev -

Customer Name: _____

Date: _____

ELEVATION



L-3 WESCAM 649 NORTH SERVICE ROAD W. BURLINGTON ONTARIO CANADA L7P 5B9
Phone : 905-633-4000 Fax : 905-633-4100

Sheet 1 of 2

Approval : _____

Customer Signature



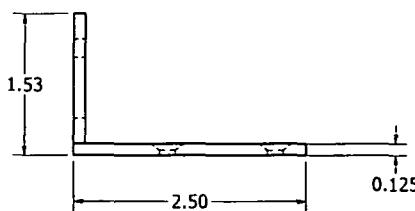
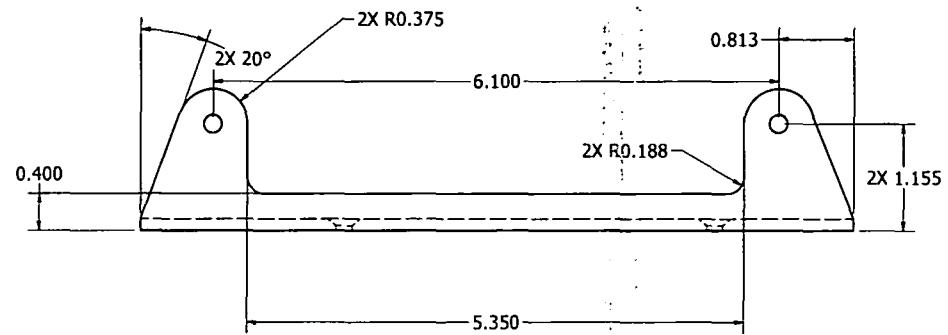
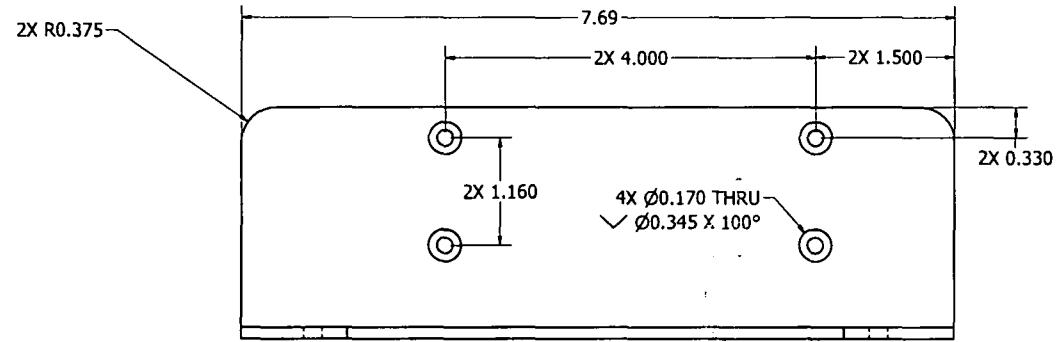
communications
WESCAM

LASER CUTOUT DEFINITION FORM

FM1287
Rev -

NOTES:

1. FOR THE FOLLOWING SURVEILLANCE SYSTEMS,
A MAXIMUM OF 9 RECTANGULAR CUTOUTS ZONES PER MAP ARE ALLOWED :
ANALOG SYSTEMS: MX15, MX20, MX15i,
DIGITAL SYSTEMS: MX15HDI
2. FOR DESIGNATOR SYSTEMS, THE LASERS ARE CUTOUT
BEYOND -85 DEGREES LOOKBACK ANGLES BY DEFAULT
3. ALTHOUGH THE GRAPH SHOWS 5 DEGREE
RESOLUTION IN AZIMUTH AND ELEVATION,
THE LASER CUTOUT CAN BE DEFINED TO 1 DEG RESOLUTION
CUSTOMER TO CLEARLY IDENTIFY THE EXACT CUTOUT
ANGLES VALUES REQUIRED, (CORNERS OF THE RECTANGULAR CUTOUTS)



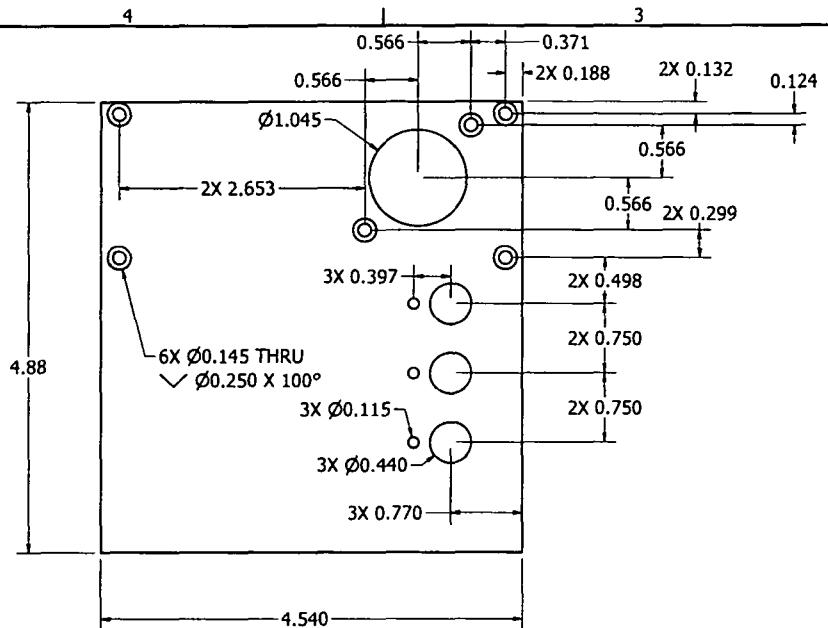
C206ARS-1000-21
MATERIAL: 6061-T6
FINISH: BLACK POLYESTER POWDER COAT CARDINAL P/N: C241-BK303

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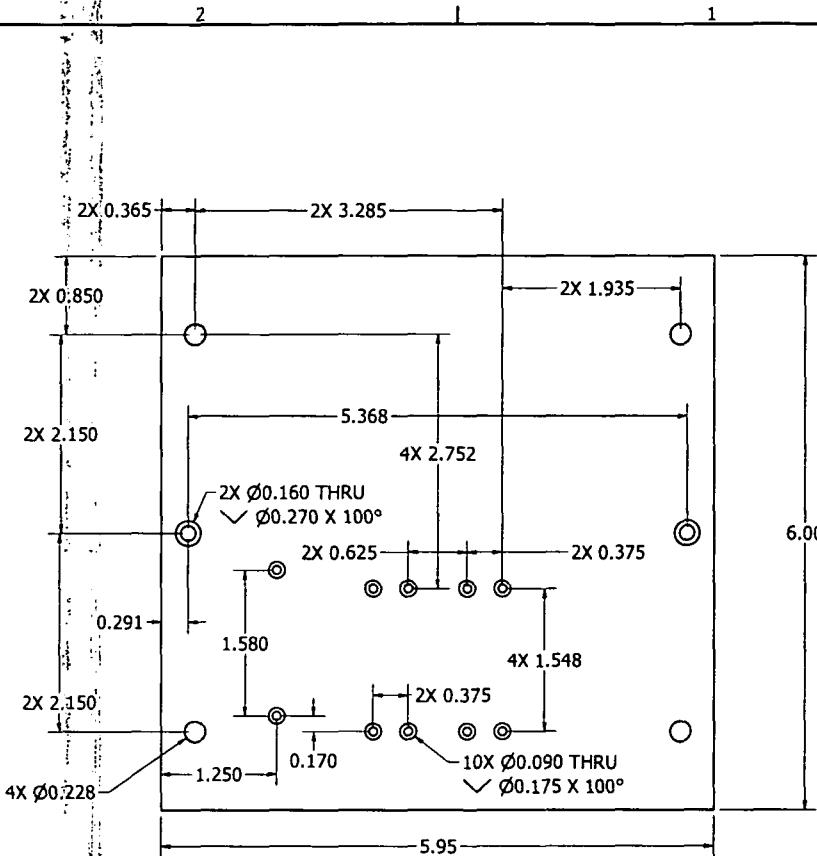
Paravion® Technology		TITLE: C206 CHURCHILL ARS EQUIPMENT		DRAWING NUMBER C206ARS-1000	
DRAWN BY	CHECKED BY	APPROVED BY	DATE	REV	ECO
HW			2/28/13	A	
DGW					

DO NOT SCALE DRAWING SHEET 1 OF 12

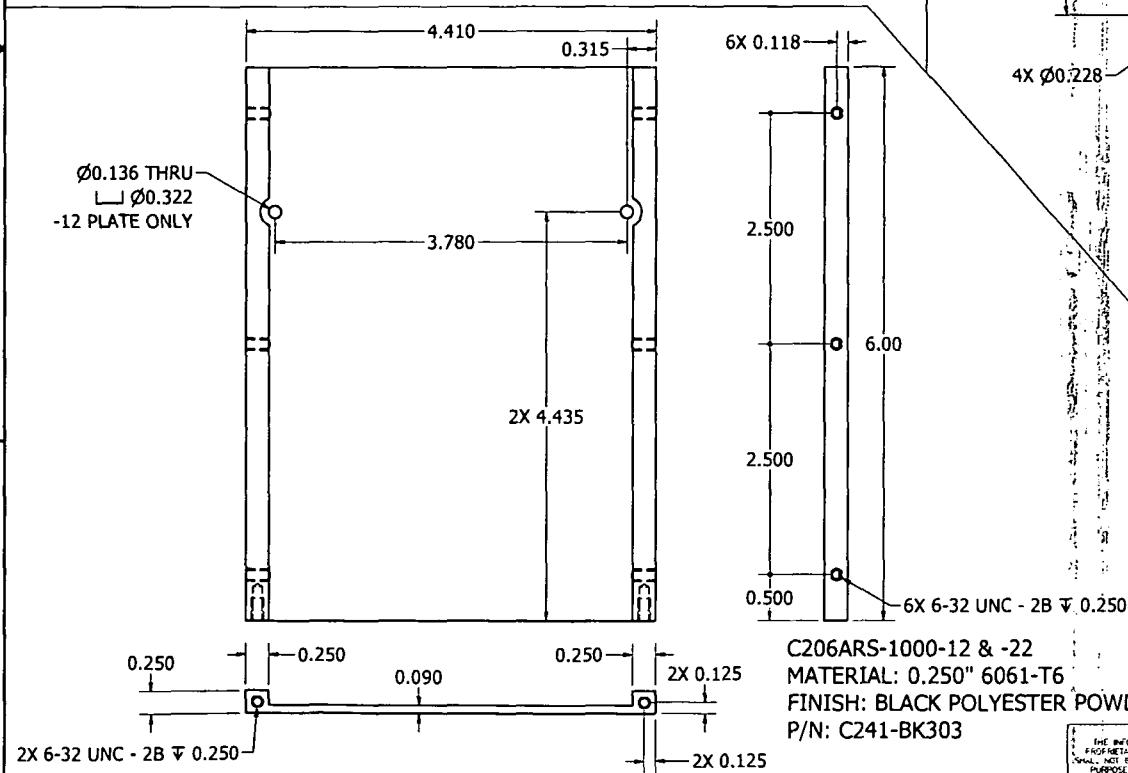
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C206ARS-1000-11
MATERIAL: 0.063" 6061-T6
FINISH: BLACK POLYESTER POWDER COAT CARDINAL P/N: C241-BK303



C206ARS-1000-10
MATERIAL: 0.063" 6061-T6
FINISH: BLACK POLYESTER POWDER COAT CARDINAL
P/N: C241-BK303



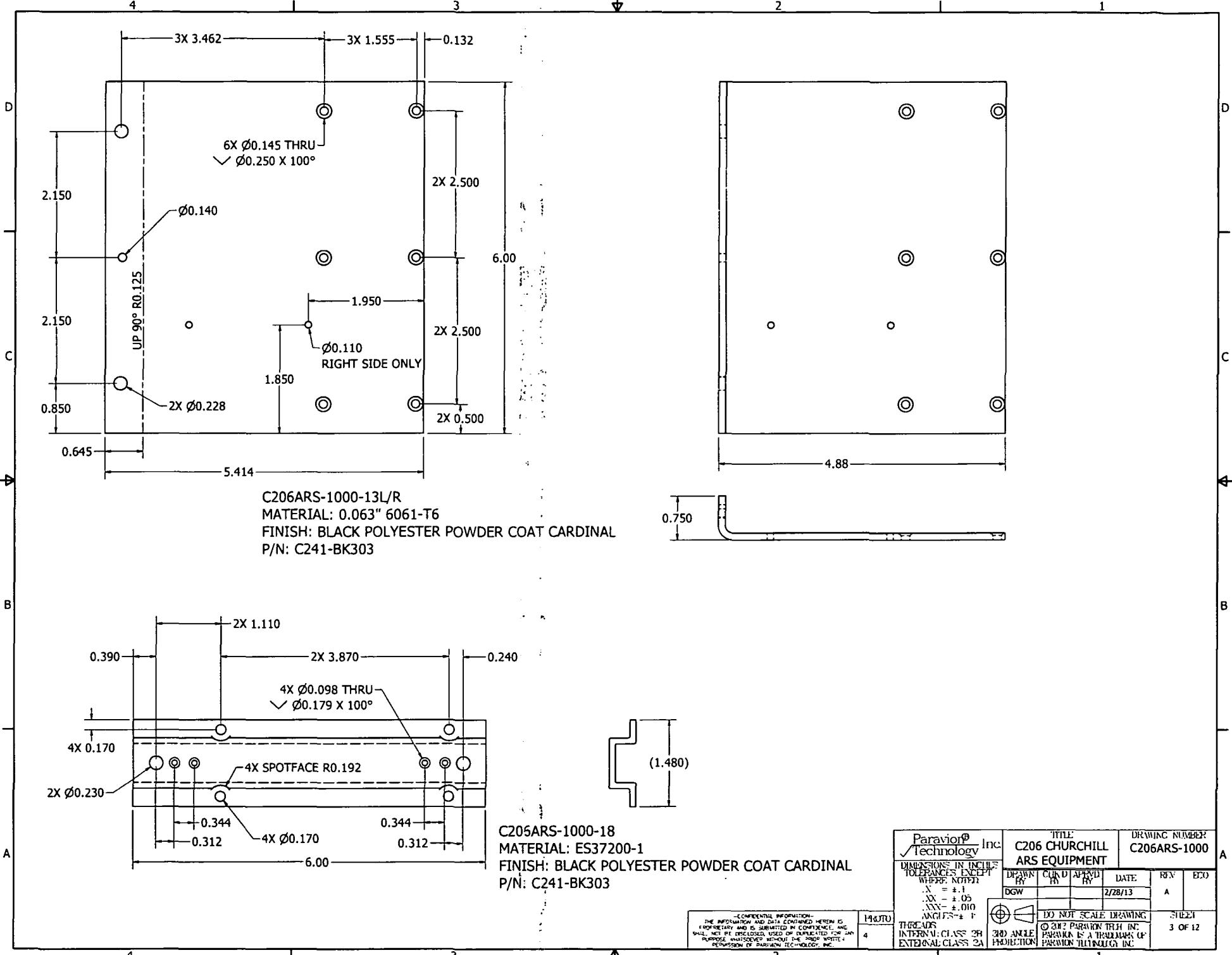
C206ARS-1000-12 & -22
MATERIAL: 0.250" 6061-T6
FINISH: BLACK POLYESTER POWDER COAT CARDINAL
P/N: C241-BK303

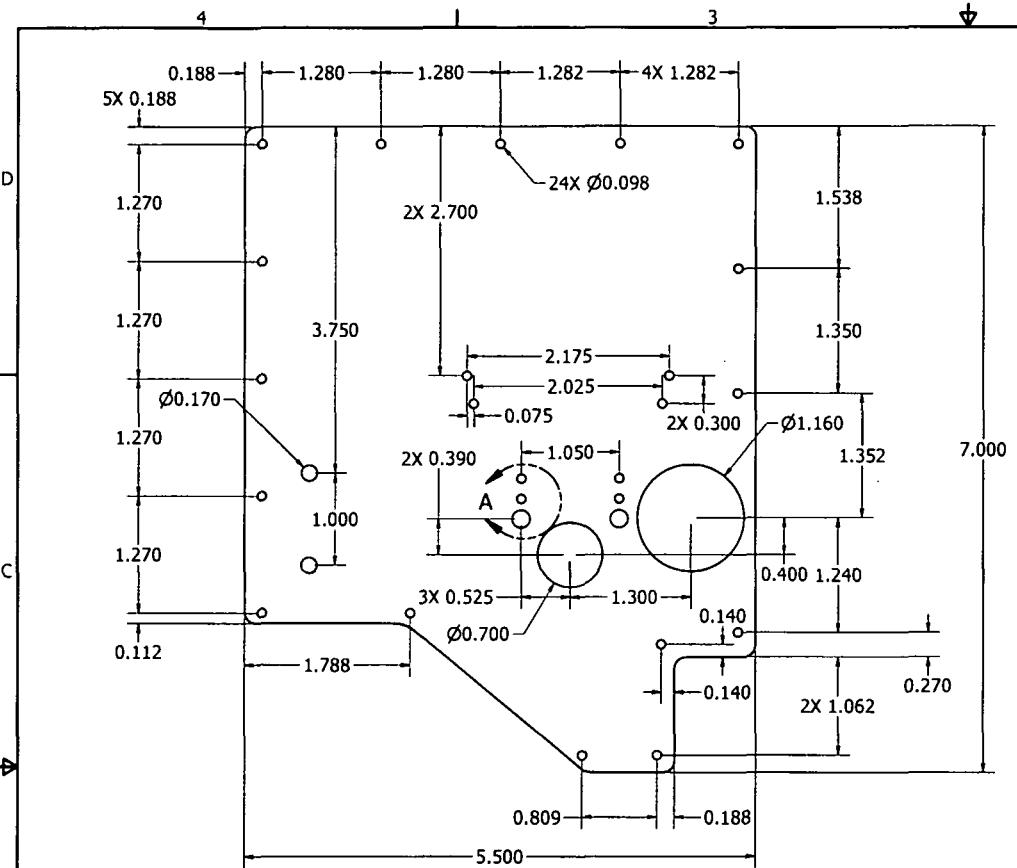
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4 M4U

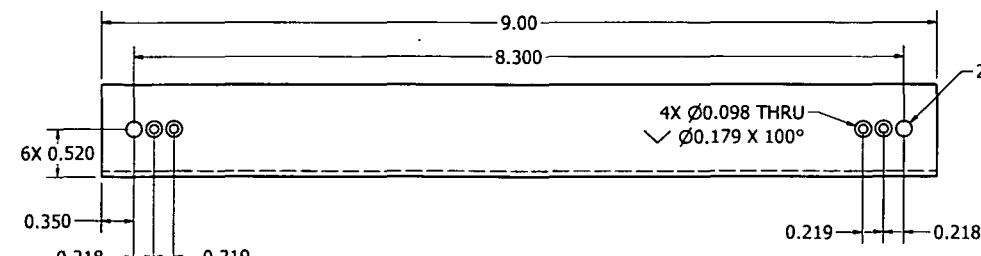
BREAK ALL SHARP EDGES

Paravion® Inc. Technology		TITLE: C206 CHURCHILL ARS EQUIPMENT		DRAWING NUMBER C206ARS-1000	
DIMENSIONS IN INCHES TOLERANCES EXCEPT WHERE NOTED:		DRAWN BY	CHECKED BY	APPROVED BY	DATE 2/28/13
XX = ± .1 XXX = ± .010 ANGLES ± 1°					REV. A ECO
THREADS INTERNAL: CLASS 2B EXTERNAL: CLASS 2A	3D ANGLE PROJECTION	M4U			
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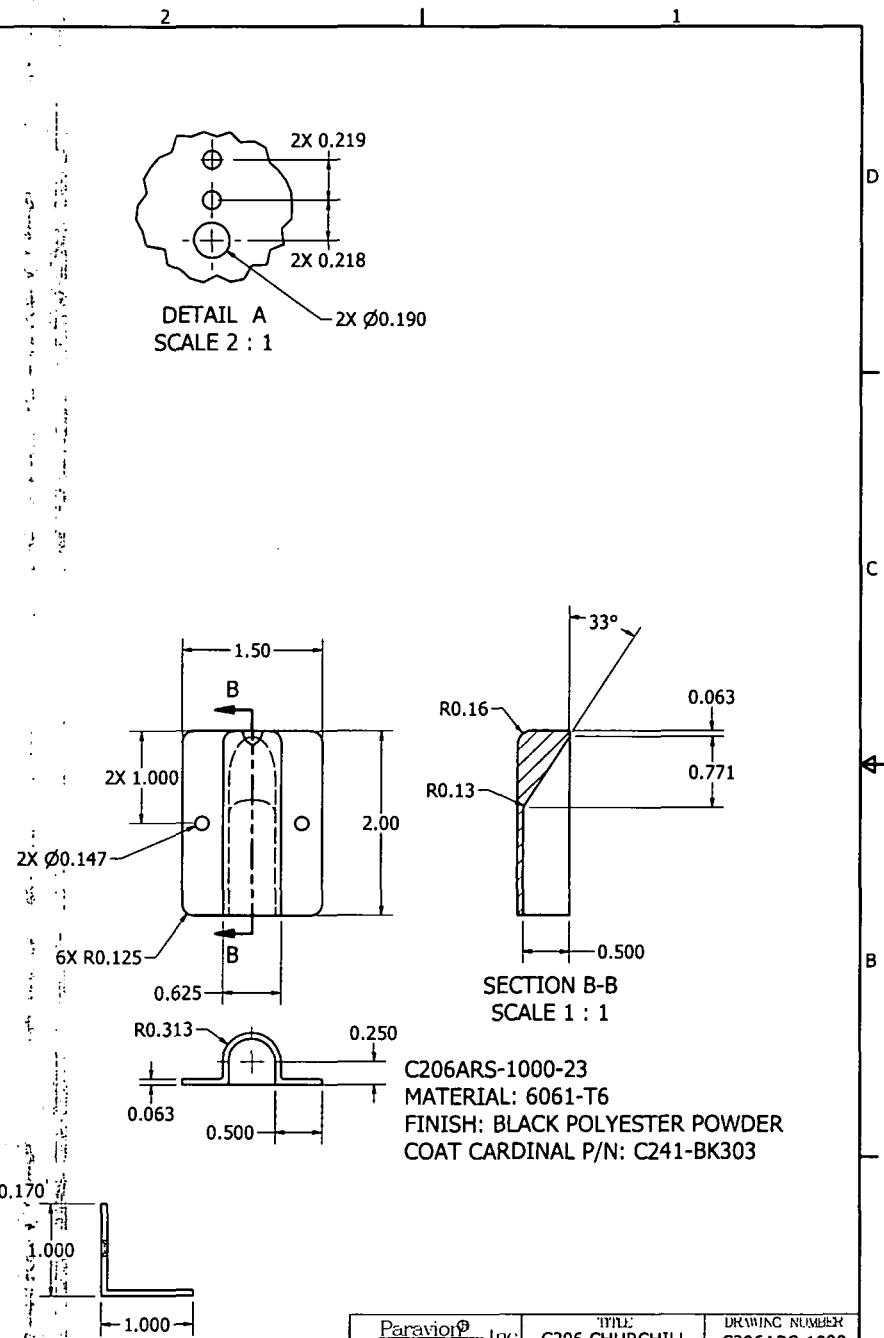




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FINISH: BLACK POLYESTER POWDER COAT CARDINAL
P/N: C241-BK303

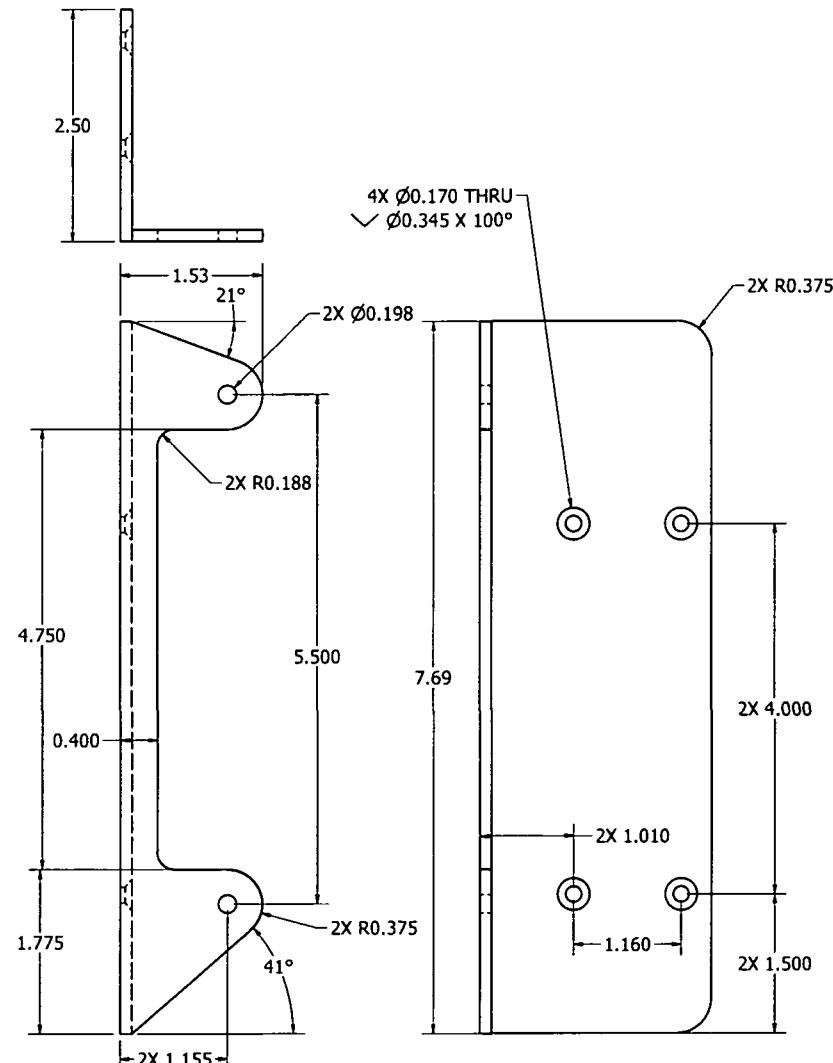
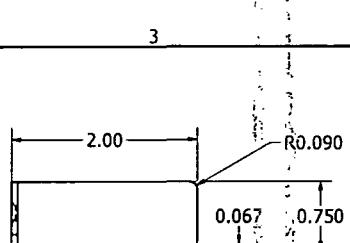
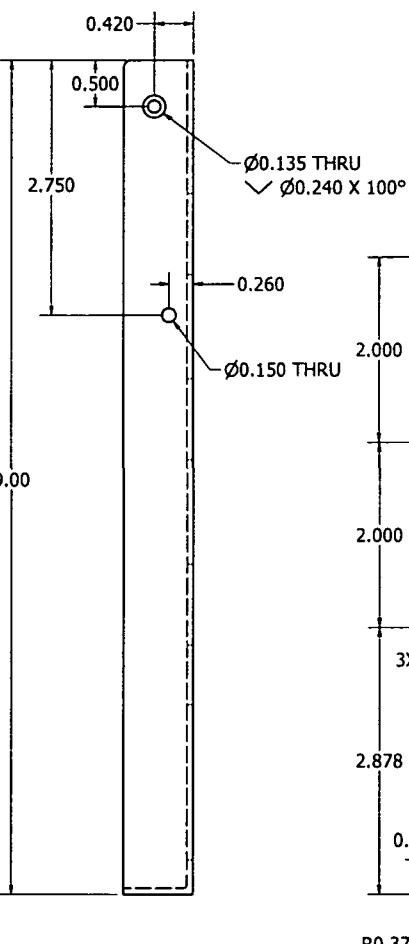


C206ARS-1000-17
MATERIAL: 1" X 1" X 0.063" ANGLE 6063-T52
FINISH: BLACK POLYESTER POWDER COAT CARDINAL
P/N: C241-BK303



C206ARS-1000-23
MATERIAL: 6061-T6
FINISH: BLACK POLYESTER POWDER COAT CARDINAL P/N: C241-BK303

Paravion® Technology		TITLE		DRAWING NUMBER	
		C206 CHURCHILL ARS EQUIPMENT		C206ARS-1000	
DIMENSIONS IN INCHES		DRAWN BY	CURR DRAFT BY	APPROV BY	DATE
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2/28/13					
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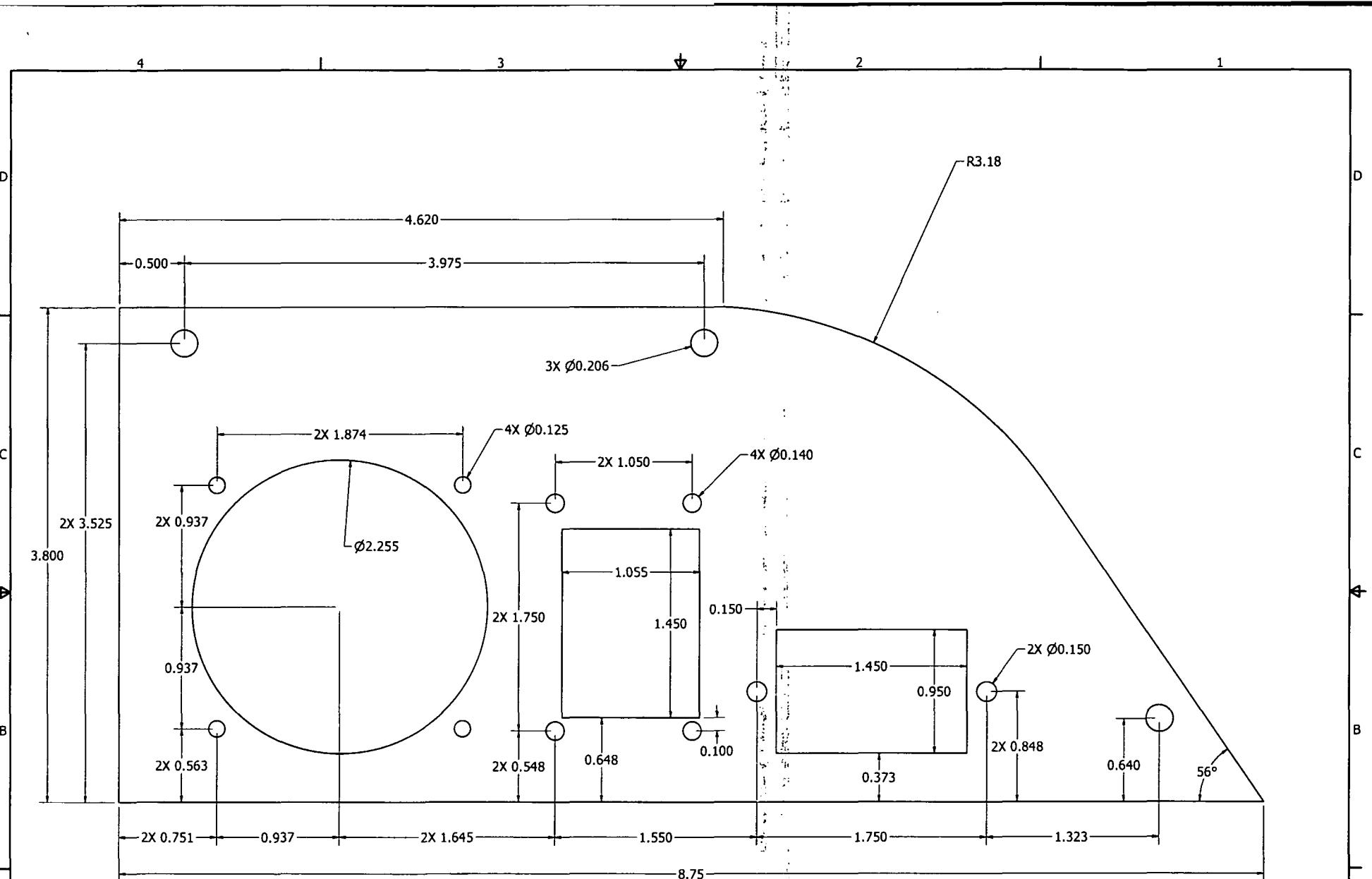


C206ARS-1000-16
MATERIAL: 6061-T6 0.125" THICK ANGLE
FINISH: BLACK POLYESTER POWDER COAT CARDINAL
P/N: C241-BK303

C206ARS-1000-15, -24
MATERIAL: 6061-T6
FINISH: BLACK POLYESTER POWDER COAT CARDINAL
P/N: C241-BK303
C206ARS-1000-19 MIRROR IMAGE

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Paravion® Technology Inc		TITLE		DRAWING NUMBER	
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DIMENSIONS IN INCHES TOLERANCES IN INCHES EXCEPT WHERE NOTED		DRAWN BY	CHECKED BY	APPROVED BY	DATE
X1 = ± .1					
X2 = ± .05					
X3 = ± .010					
X4 = ± .1					
THREADS		DO NOT SCALE DRAWING		SHEET	
INTERNAL CLASS 2B EXTERNAL CLASS 2A		© 2012 PARAVION TECH, INC. PARAVION IS A TRADEMARK OF PARAVION TECHNOLOGY, INC.		5 OF 12	
3RD ANGLE PROJECTION					



C206ARS-1000-20
MATERIAL: 6061-T6 0.063"
FINISH: BLACK POLYESTER POWDER COAT CARDINAL
P/N: C241-BK303

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Paravion® Inc.		TITLE		DRAWING NUMBER	
		C206 CHURCHILL ARS EQUIPMENT		C206ARS-1000	
DIMENSIONS IN INCHES		DRAWN BY	CHECKED BY	APPROVED BY	DATE
NOTES:					
W.H.P. - NOV 12	X = ± .1				2/28/13
	.XX = ± .05				
	.XXX = ± .010				
	ANGLES = ± 1°				
THREADS:		DO NOT SCALE DRAWING		SHEET	
INTERNAL: CLASS 2B		© 2012 PARAVION TECH, INC.		6 OF 12	
EXTERNAL: CLASS 2A		PARAVION IS A TRADEMARK OF PARAVION TECHNOLOGY, INC.			
DRAW ANGLE PROJECTION					

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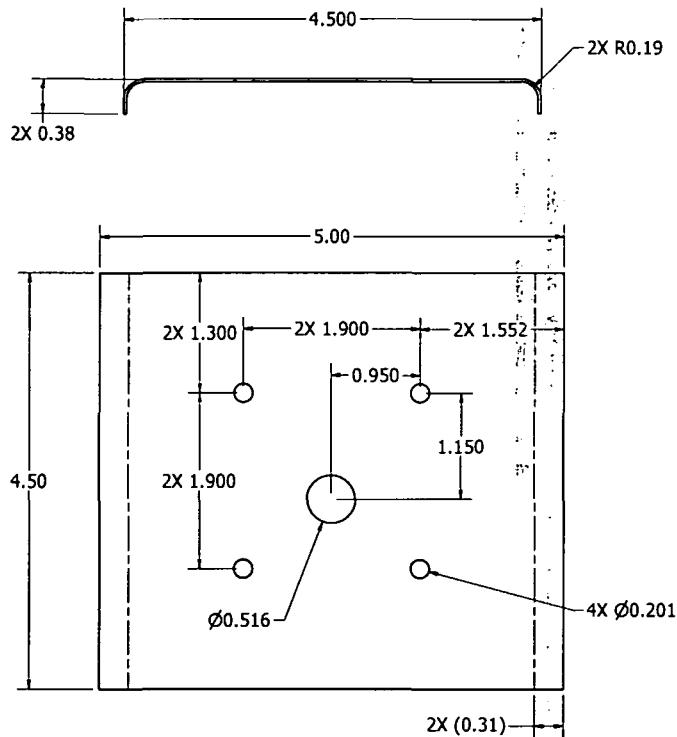
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C206ARS-1000-25

MATERIAL: 6061-T6 0.032"

FINISH: BLACK POLYESTER POWDER COAT CARDINAL

P/N: C241-BK303

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4

PHOTO

Paravion [®] Inc		TITLE		DRAWING NUMBER	
Technology		C206 CHURCHILL		C206ARS-1000	
ARS EQUIPMENT					
DRAWN	CUR.D	APPROV'D	DATE	REV	ECD
RT	HT	HT		A	
DGW				2/28/13	



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7 OF 12

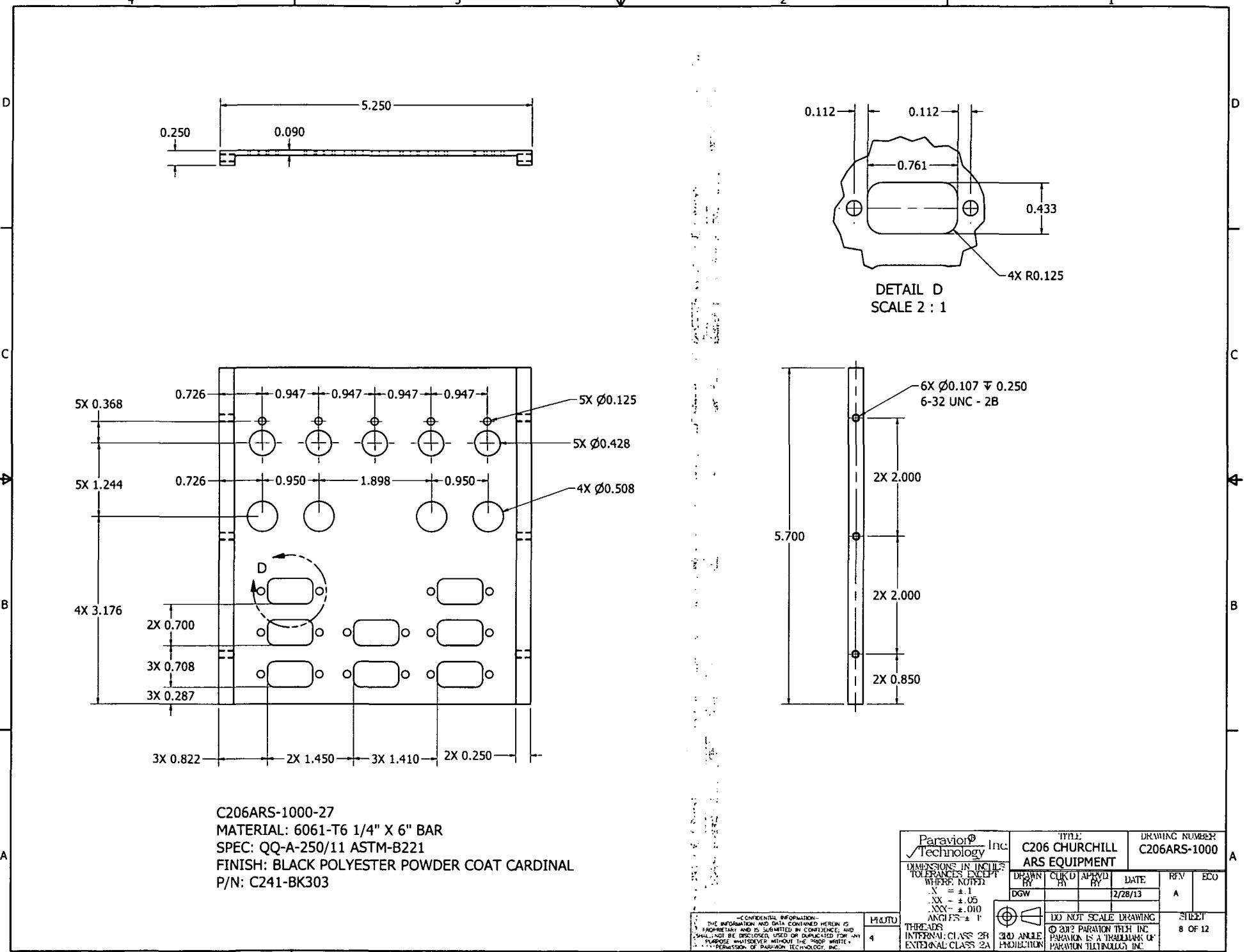
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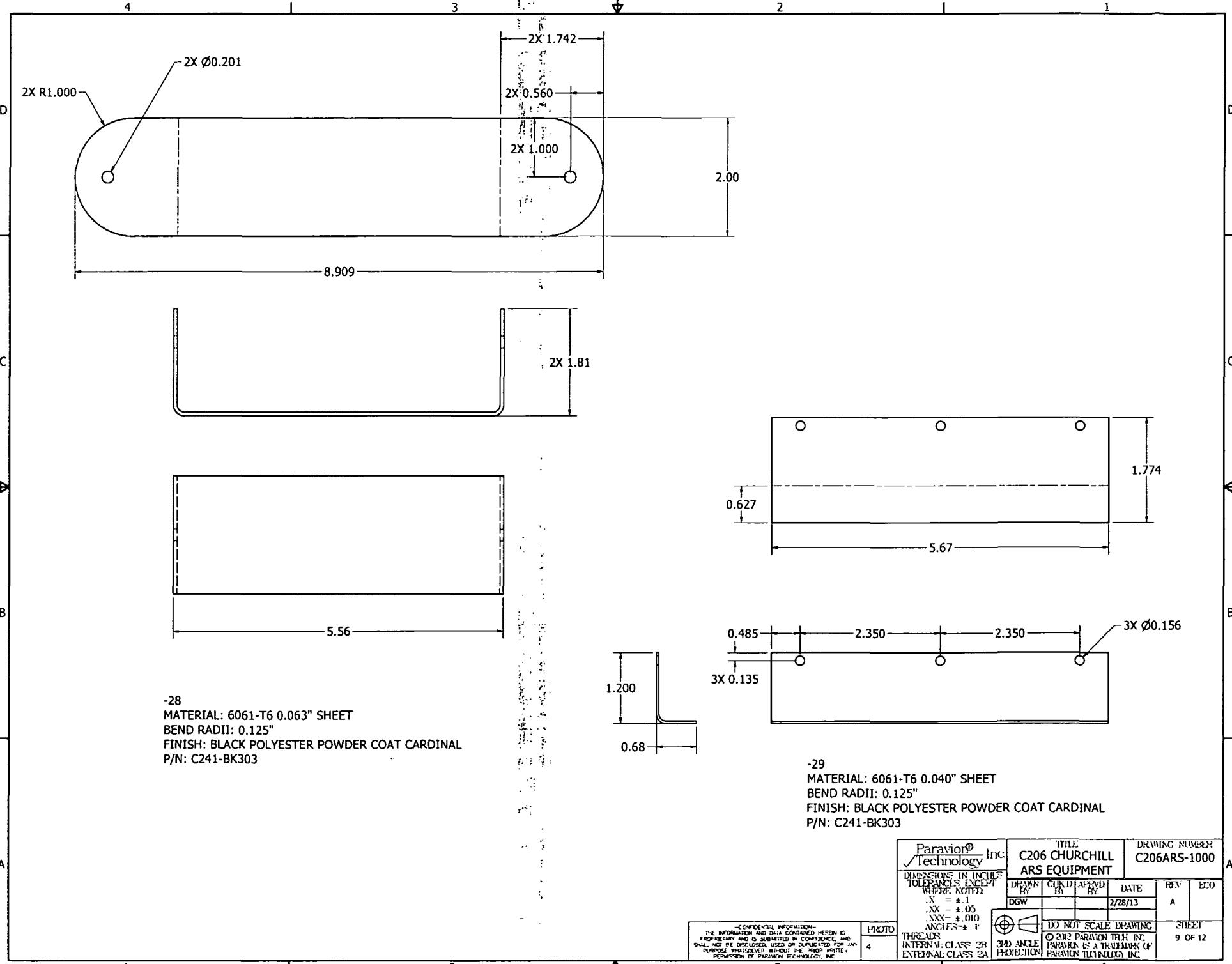
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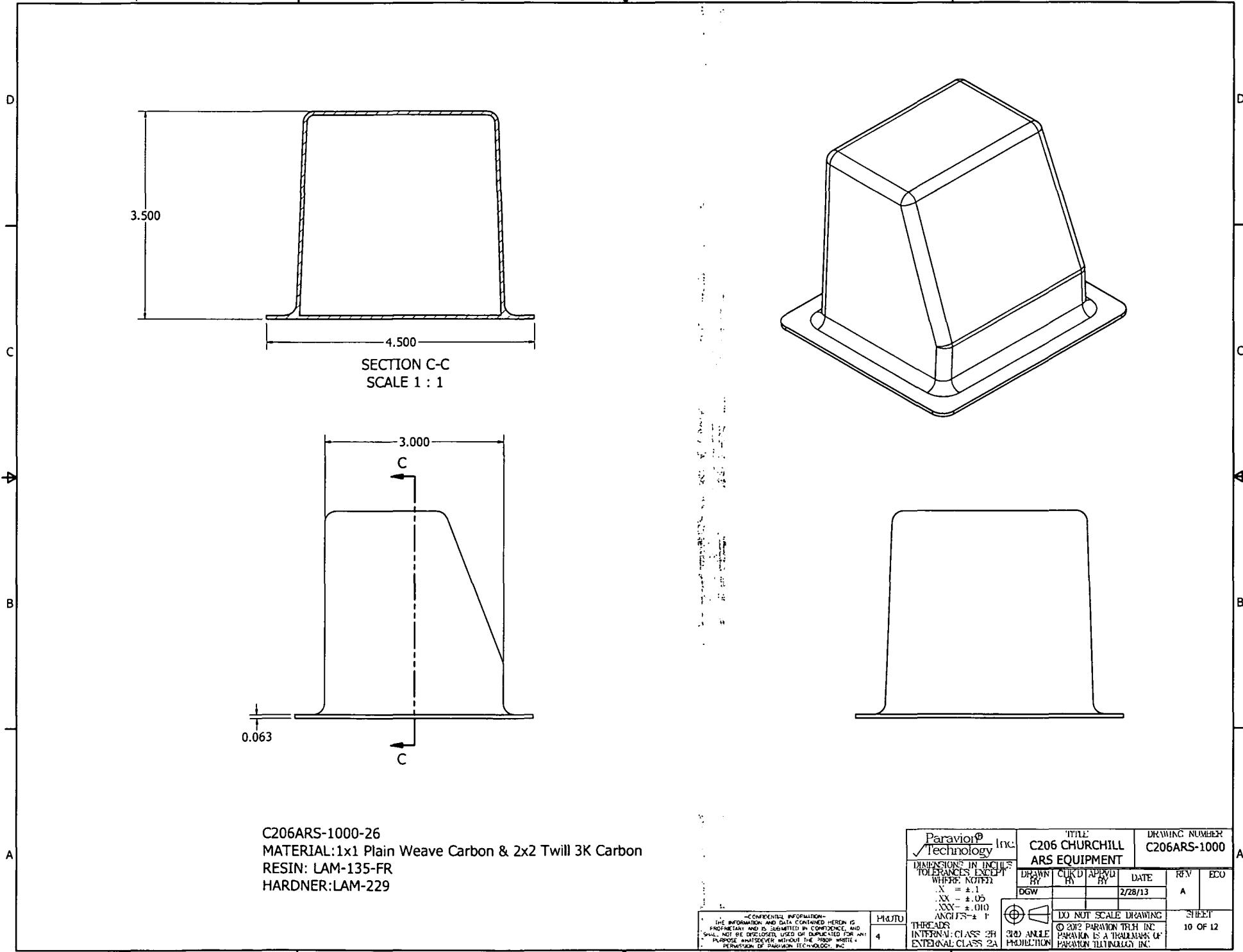
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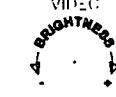
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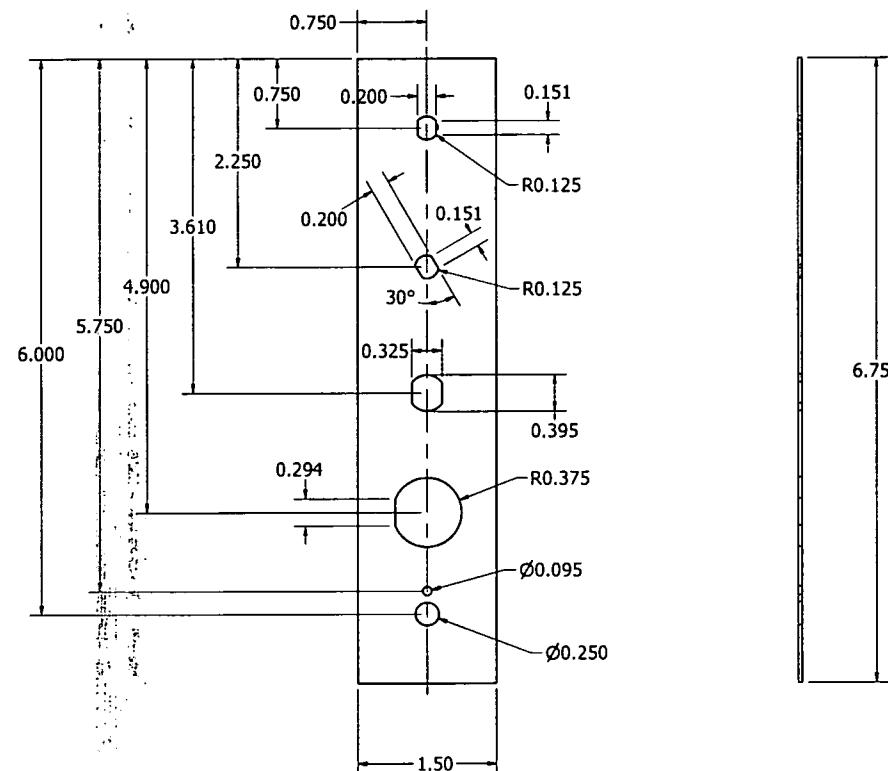
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-31

MATERIAL: 6061-T6 0.040" SHEET
FINISH: BLACK POLYESTER POWDER COAT CARDINAL
P/N: C241-BK303

Paravion® Technology Inc.		TITLE: C206 CHURCHILL ARS EQUIPMENT			DRAWING NUMBER: C206ARS-1000		
		DRAWN BY: DGW	CHECKED BY: HJ	APPROVED BY: HJ	DATE: 2/28/13	REV: A	ECO:
DIMENSIONS IN INCHES TOLERANCES EXCEPT WHERE NOTED: XX = ±.1 XXX = ±.05 XXXX = ±.010 ANGLES ± 1°							
THREADS INTERNAL CLASS 3B EXTERNAL CLASS 2A							
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SHEET 11 OF 12							

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PHOTO

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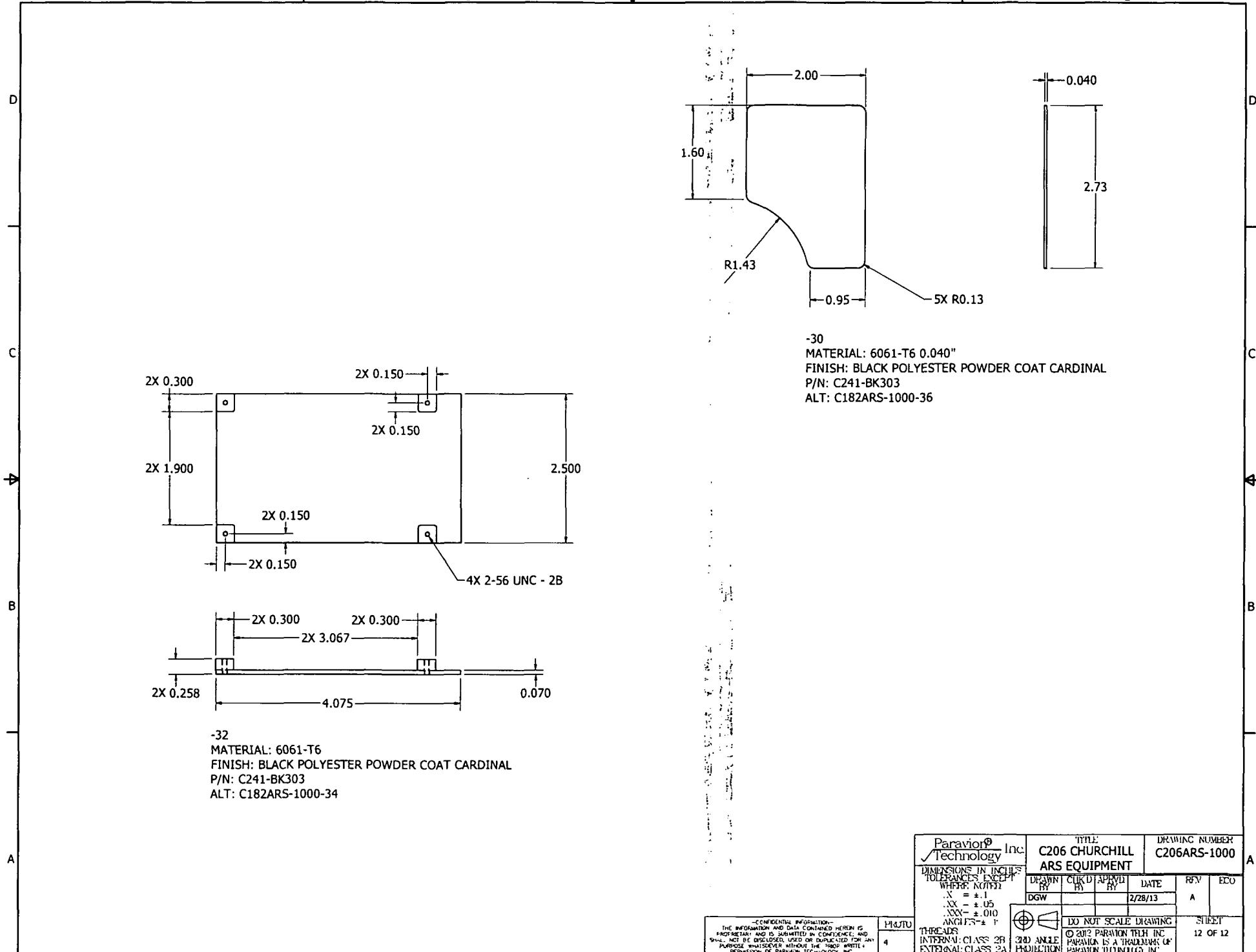
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Paravion® Inc.		TITLE: C206 CHURCHILL ARS EQUIPMENT		DRAWING NUMBER: C206ARS-1000	
DIMENSIONS IN INCHES		DRAWN BY	CHECKED BY	APPROVED BY	DATE
TOLERANCES EXCEPT WHERE NOTED					
.+-. = ±.1					
.XX = ±.05					
.XXX = ±.010					
ANGLES = ± 1°					
DO NOT SCALE DRAWING					
THREADS INTERNAL: CLASS 2B EXTERNAL: CLASS 2A					
END ANGLE PROJECTION					
PARAVION IS A TRADEMARK OF PARAVION TECHNOLOGY INC.					

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PROJ#04

4

THREADS

INTERNAL: CLASS 2B

EXTERNAL: CLASS 2A

END ANGLE

PROJECTION

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STREET

12 OF 12

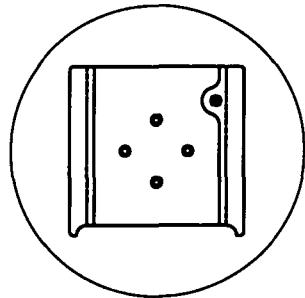
Bill of Materials

*IR-440-1, REV. N/C

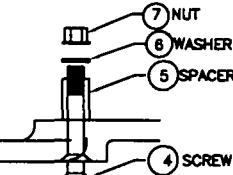
ITEM #	P/N	DESCRIPTION	QTY	TYP
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1	IR-606-1	SUPPORT PLATE	4	EA
2	MS16998-44	BOLT	4	EA
3	NAS1149C0463R	WASHER	1	EA
4	MS24693S279	SCREW	1	EA
5	NAS43DD3-32FC	SPACER	1	EA
6	NAS1149F0332P	WASHER	1	EA
7	MS21042L3	NUT	1	EA

SHEET	REV	ECO	DATE	DESCRIPTION	BY	APR	CHK

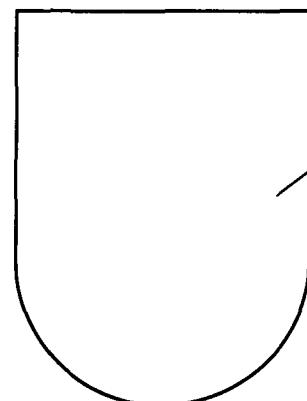
4 FORWARD



- ① SUPPORT PLATE
② CAP SCREW (TYP. 4 PLCS)
③ WASHER (TYP. 4 PLCS)



DETAIL A



WESCAM MX10 TURRET
OUTLINE SHOWN

-1 ASSEMBLY

-CONFIDENTIAL INFORMATION-
THE INFORMATION AND DATA CONTAINED HEREIN
IS PROPRIETARY AND IS SUBMITTED IN CONFIDENCE
AND SHALL NOT BE DISCLOSED, USED OR DUPLICATED
FOR ANY PURPOSE WHATSOEVER WITHOUT THE PRIOR
WRITTEN PERMISSION OF PARAVION TECHNOLOGY, INC.

PROTO

1

Paravion® Inc. Technology		TITLE GIMBAL ADAPTER			DRAWING NO. IR-440		
DIMENSIONS IN INCHES		DRAWN BY REB	APRVD. BY REB	CHK'D. BY CEH	DATE 02/17/2012	REV N/C	ECO -
TOLERANCES EXCEPT WHERE NOTED:		.X = ± .1	.XX = ± .05	.XXX = ± .010	ANGLES = ± 1°	DO NOT SCALE DRAWING	
© 2008 PARAVION TECH., INC. PARAVION IS A TRADEMARK OF PARAVION TECHNOLOGY, INC.						SHEET 1 OF 1	

PARAVION TECHNOLOGY, INC	Instructions for Continued Airworthiness IR-605-1 Support Plate – Cessna 206		
	Revision: <u>IR</u>	Date: <u>12/19/2012</u>	A/C N#: _____ A/C S/N: <u>T20608983</u>

The installation is to be inspected in accordance with the following criteria or equivalent operator's Approved Airworthiness Inspection Program:

1.0 INTRODUCTION

These Instructions for Continued Airworthiness contain the necessary information for carrying out the ongoing maintenance and inspections on the installation of an IR-605-1 Support Plate on a Cessna 206 aircraft in accordance with FAA Form 337 dated _____.

2.0 DESCRIPTION

Paravion Technology drawing IR-605 describes the support plate used as provisions for a Wecam MX-10 camera. The plate is installed in the belly of the aircraft at approximately the center of the fuselage along the centerline of the aircraft. It weighs approximately 1.6 lbs and is installed using 4 x NAS1351C4 screws.

3.0 CONTROL, OPERATION INFORMATION

N/A

4.0 SERVICING INFORMATION

N/A

5.0 MAINTENANCE INSTRUCTIONS

The inspection program for this installation consists of a 12-month annual inspection for the condition of the support plate and associated components. This inspection is a complete visual inspection requiring only a single logbook entry.

12-Month Inspection

- A. Inspect condition of support plate and all associated mounting structure for loose hardware or damage, i.e. bent, cracked or dented structures, and repair or replace as necessary.

The 12-month inspections shall be accomplished by an appropriately rated mechanic assigned to this aircraft and can be accomplished earlier to match up with other aircraft inspections.

6.0 TROUBLESHOOTING

N/A

7.0 REMOVAL AND REPLACEMENT INSTRUCTIONS

- A. Paravion Technology drawing IR-605 (provided) shows the details of the support plate and report number ER-C206ELP-2 shows the installation of the plate and can be used as a reference in the event the plate needs to be removed and replaced.

8.0 DIAGRAMS

N/A

PARAVION TECHNOLOGY, INC	Instructions for Continued Airworthiness IR-605-1 Support Plate – Cessna 206 Revision: <u>IR</u> Date: <u>12/19/2012</u> A/C N#: _____ A/C S/N: <u>T20608983</u>		

9.0 SPECIAL INSPECTION REQUIREMENTS
N/A

10.0 APPLICATION OF PROTECTIVE TREATMENTS
N/A

11.0 STRUCTURAL DATA
N/A

12.0 LIST OF SPECIAL TOOLS
N/A

13.0 FOR COMMUTER CATEGORY AIRCRAFT
N/A

14.0 RECOMMENDED OVERHAUL INTERVALES
No additional overhaul time limitations.

15.0 AIRWORTHINESS LIMITATIONS

The Airworthiness Limitations section is FAA approved and specifies inspections and other maintenance required under §43.16 and §91.403 of the Federal Aviation Regulations unless an alternative program has been FAA approved.
No additional airworthiness limitations.

16.0 REVISIONS

ICAs are required to be acceptable to the FAA. As such, changes should be documented by submitting the revised ICA along with the original Form 337 to the Aircraft Registration Branch in Oklahoma City. An entry in the aircraft records should indicate the current revision.

***** NOTHING FOLLOWS *****



U.S. Department of
Transportation
Federal Aviation
Administration

**MAJOR REPAIR AND ALTERATION
(Airframe, Powerplant, Propeller, or Appliance)**

Form Approved
OMB No. 2120-0020
11/30/2007

Electronic Tracking Number

For FAA Use Only

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act 1958).

1. Aircraft	Nationality and Registration Mark USA N959JT	Serial No. T20608983
	Make CESSNA	Model T206H
2. Owner	Name (As shown on registration certificate) CESSNA AIRCRAFT COMPANY	Address (As shown on registration certificate) Address ATTN: DEPT 093 3 CESSNA BLVD City WICHITA State KANSAS Zip 67215-1400 Country USA

3. For FAA Use Only

4. Type		5. Unit Identification			
Repair	Alteration	Unit	Make	Model	Serial Number
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in Item 1 above)	_____
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT	_____	_____	_____
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER	_____	_____	_____
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type	_____	_____
<input type="checkbox"/>	<input type="checkbox"/>		Manufacturer		

6. Conformity Statement

A. Agency's Name and Address		B. Kind of Agency
Name YINGLING AVIATION Address 2010 AIRPORT ROAD City WICHITA State KS Zip 67277 Country USA		<input type="checkbox"/> U.S. Certificated Mechanic <input type="checkbox"/> Manufacturer <input type="checkbox"/> Foreign Certificated Mechanic <input type="checkbox"/> Certificate No. <input checked="" type="checkbox"/> Certificated Repair Station YN8R621Y <input type="checkbox"/> Certificated Maintenance Organization RADIO CLASS 1,2,3

D. I certify that the repair and/or alteration made to the unit(s) identified in Item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

Extended range fuel per 14 CFR Part 43 App. B	<input type="checkbox"/>	Signature/Date of Authorized Individual <i>Aug 11/10</i>
--	--------------------------	---

7. Approval for Return to Service

Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED REJECTED

BY	FAA Fit Standards Inspector	Manufacturer	Maintenance Organization	Person Approved by Canadian Department of Transport
	FAA Designee	<input checked="" type="checkbox"/> Repair Station	Inspection Authorization	Other (Specify)
Certificate or Designation No. YN8R621Y		Signature/Date of Authorized Individual <i>HC dlr 11/11/10</i>		

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

USA N959JT

Nationality and Registration Mark

Date

CESSNA T206H - T20608983- N959JT

INSTALLED SYSTEMS: The following avionics equipment was installed IAW STC Number SA01552WI. Reference Yingling Aviation STC Master Data List, Document No. 23507-F206-M Revision IR dated August 3, 2009 or later approved revision.

Garmin GMA 1347 Audio Panel, 2 ea NAT (A711) Expansion Panels, NAT Audio Control System (A740), NAT Universal Audio Radio Interface, 3 ea Sandia Card Enclosures (SRU-1), 3 ea Sandia Relay Cards (SR-54), KGS Model RG28 DC to DC Converter, GPS Antenna CI-420-230, Marker Beacon Antenna Splitter CI509, and 2 ea Comant VHF/FM CI-292-3 Antennas.

OPERATIONAL GROUND CHECKS: Required ground tests were performed and all equipment was found to operate properly.

CONTINUED AIRWORTHINESS INSTRUCTIONS: Reference Yingling Aviation Document ICA 23507-F206-04 for Instructions for Continued Airworthiness.

WEIGHT & BALANCE and EQUIPMENT LIST: Revised Aircraft Weight & Balance and Equipment List. See Aircraft Weight & Balance records for details.

AFMS: FAA Approved Flight Manual Supplement Doc. No. AFMS Document 23507-F206-08, Rev.IR, dated August 12, 2009 inserted in the Airplane Flight Manual.

The above installation meets the requirements for static loading in accordance with A.C.43.13-2B Chapter 1 par. 106 through 114. No changes were noted to the compass system. Further details are on file at C.R.S. # YN8R621Y under W.O. # AVI 10096.

>>>>>>>>>>END<<<<<<<<<<<

Additional Sheets Are Attached



U.S Department of
Transportation
Federal Aviation
Administration

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved
OMB No. 2120-0020
11/30/2007

Electronic Tracking Number

For FAA Use Only

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation / Section 901 Federal Aviation Act 1958.

1. Aircraft	Nationality and Registration Mark USA N959JT	Serial No. T20608983
	Make CESSNA	Model T206H
2. Owner	Name (As shown on registration certificate) CESSNA AIRCRAFT COMPANY	Address (As shown on registration certificate) Address ATTN: DEPT 093 3CESSNA BLVD City WICHITA State KANSAS Zip 67215-1400 Country USA

3. For FAA Use Only

4. Type		5. Unit Identification		
Repair	Alteration	Unit	Make	Model
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in Item 1 above)
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT	_____	_____
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER	_____	_____
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type	_____
<input type="checkbox"/>	<input type="checkbox"/>		Manufacturer	_____

6. Conformity Statement

A. Agency's Name and Address		B. Kind of Agency
Name YINGLING AVIATION Address 2010 AIRPORT ROAD City WICHITA State KS Zip 67277 Country USA		<input type="checkbox"/> U.S. Certificated Mechanic <input type="checkbox"/> Manufacturer <input type="checkbox"/> Foreign Certificated Mechanic C. Certificate No. <input checked="" type="checkbox"/> Certificated Repair Station YN8R621Y <input type="checkbox"/> Certificated Maintenance Organization RADIO CLASS 1,2,3

D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

Extended range fuel per 14 CFR Part 43 App. B	<input type="checkbox"/>	Signature/Date of Authorized Individual <i>Deepley Jones Jr 11/11/10</i>
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7. Approval for Return to Service

Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED REJECTED

BY	FAA Fit Standards Inspector	Manufacturer	Maintenance Organization	Person Approved by Canadian Department of Transport
	FAA Designee	X Repair Station	Inspection Authorization	Other (Specify)
Certificate or Designation No. YN8R621Y		Signature/Date of Authorized Individual <i>H.C. Sherd 11/11/10</i>		

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

USA N959JT

11/11/10

Nationality and Registration Mark

Date

CESSNA T206H- T20608983- N959JT

Installed Equipment: The following equipment was installed IAW Atlantic Aero STC SA03150AT-D. Installation of a Wulfsberg P-2000 FM Radio in accordance with Atlantic Aero Master Drawing List 24293001 Rev B dated 6/4/08.

Wulfsberg P-2000VHF Digital/Analog Tactical FM Radio:

- Wulfsberg FM Transceiver P-2000VHF P/N 400-049200-11-011-2135-2135, 3.7 lbs @ arm 15.3
- Comant VHF FM (bent whip) Antenna P/N CI292-3, .5 lbs @ arm 55.3
- Comant VHF Antenna P/N CI 177-1, .5 lbs @ arm 155.8

OPERATIONAL GROUND CHECKS: Post installation ground functional and interference tests were performed IAW Atlantic Aero Doc 24291001.

CONTINUED AIRWORTHINESS INSTRUCTIONS: Reference Atlantic Aero ICA24293010 Rev A dated 6/25/2008 for Instructions for Continued Airworthiness.

WEIGHT & BALANCE and EQUIPMENT LIST: Revised Aircraft Weight & Balance and Equipment List. See Aircraft Weight and Balance records for details.

AFMS: Atlantic Aero FAA Approved Flight Manual Supplement Doc. No. DAS-511128-CE, Rev. A, dated 6/25/08 inserted in the Airplane Flight Manual.

The above installation meets the requirements for static loading in accordance with A.C.43.13-2B Chapter 1 par. 106 through 114. No changes were noted to the compass system. Further details are on file at C.R.S. # YN8R621Y under W.O. # AVI 10096.

>>>>>>>>>>END<<<<<<<<<<<

Additional Sheets Are Attached



U.S. Department of
Transportation
Federal Aviation
Administration

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved
OMB No. 2120-0020
11/30/2007

Electronic Tracking Number

For FAA Use Only

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act 1958).

1. Aircraft	Nationality and Registration Mark USA N959JT		Serial No. T20608983	
	Make CESSNA		Model T206H	Series STATIONAIR
2. Owner	Name (As shown on registration certificate) CESSNA AIRCRAFT COMPANY		Address (As shown on registration certificate) Address ATTN: DEPT 093 3 CESSNA BLVD City WICHITA Zip 67215-1400 Country USA	
3. For FAA Use Only.				

4. Type		5. Unit Identification			
Repair	Alteration	Unit	Make	Model	Serial Number
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in Item 1 above)	
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT	_____		
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER	_____		
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type Manufacturer		

6. Conformity Statement		
A. Agency's Name and Address		B. Kind of Agency
Name YINGLING AVIATION Address 2010 AIRPORT ROAD City WICHITA Zip 67277 Country USA		<input type="checkbox"/> U.S. Certificated Mechanic <input type="checkbox"/> Foreign Certificated Mechanic <input checked="" type="checkbox"/> Certificated Repair Station <input type="checkbox"/> Certificated Maintenance Organization
		<input type="checkbox"/> Manufacturer C. Certificate No. YN8R621Y RADIO CLASS 1,2,3

D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.		
Extended range fuel per 14 CFR Part 43 App. B	<input type="checkbox"/>	Signature/Date of Authorized Individual <i>Leozilo Sines Jr 11/11/10</i>
7. Approval for Return to Service		

Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> APPROVED <input type="checkbox"/> REJECTED					
BY	FAA Fit Standards Inspector	Manufacturer	Maintenance Organization	Person Approved by Canadian Department of Transport	
	FAA Designee	X Repair Station	Inspection Authorization	Other (Specify)	
Certificate or Designation No. YN8R621Y		Signature/Date of Authorized Individual <i>John C. de la Torre 11/11/10</i>			

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

USA N959JT

11/11/10

Nationality and Registration Mark

Date

CESSNA T206H – T20608983– N959JT

INSTALLED SYSTEMS: Installed Precise Flight Pulselite Control unit IAW STC SA4005NM.

OPERATIONAL GROUND CHECKS: Required ground tests were performed and the equipment was found to operate normally IAW Precise Flight Installation Manual PPRI-2000 Doc # 015PMAN0001 Rev.O dated May 16, 2007.

CONTINUED AIRWORTHINESS INSTRUCTIONS: Reference Document No. 000PMAN0002 Rev. D (7/07) for Instructions for Continued Airworthiness.

AFMS: FAA Approved Flight Manual Supplement Doc. No. 000PMAN0001Rev. A (7/24/03) was inserted into the Aircraft Flight Manual.

WEIGHT & BALANCE and EQUIPMENT LIST: Revised Aircraft Weight & Balance and Equipment List. See Aircraft Weight & Balance records for details.

The above installation meets the requirements for static loading in accordance with A.C.43.13-2B Chapter 1 par. 106 through 114. No changes were noted to the compass system. Further details are on file at C.R.S. # YN8R621Y under W.O. # AVI 10096.

>>>>>>>>>END<<<<<<<<<

Additional Sheets Are Attached



U.S. Department of
Transportation
Federal Aviation
Administration

MAJOR REPAIR AND ALTERATION
(Airframe, Powerplant, Propeller, or Appliance)

Form Approved
OMB No. 2120-0020
11/30/2007

Electronic Tracking Number

For FAA Use Only

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act 1958)

1. Aircraft	Nationality and Registration Mark USA N959JT	Serial No. T20608983	
	Make CESSNA	Model T206H	Series STATIONAIR
2. Owner	Name (As shown on registration certificate) CESSNA AIRCRAFT COMPANY	Address (As shown on registration certificate) Address ATTN: DEPT 093 3 CESSNA BLVD City WICHITA Zip 67215-1400	State KANSAS Country USA

3. For FAA Use Only

4. Type		5. Unit Identification			
Repair	Alteration	Unit	Make	Model	Serial Number
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in Item 1 above)	_____
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT	_____	_____	_____
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER	_____	_____	_____
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type _____	Manufacturer _____	_____

6. Conformity Statement

A. Agency's Name and Address		B. Kind of Agency		
Name YINGLING AVIATION Address 2010 AIRPORT ROAD City WICHITA Zip 67277		<input type="checkbox"/> U.S. Certificated Mechanic <input type="checkbox"/> Manufacturer <input type="checkbox"/> Foreign Certificated Mechanic C. Certificate No. <input checked="" type="checkbox"/> Certificated Repair Station YN8R621Y <input type="checkbox"/> Certificated Maintenance Organization RADIO CLASS 1,2,3		

D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

Extended range fuel per 14 CFR Part 43 App. B	<input type="checkbox"/>	Signature/Date of Authorized Individual <i>Delegorales Jr 11/11/10</i>		
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7. Approval for Return to Service

Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED REJECTED

BY	FAA Fit Standards Inspector	Manufacturer	Maintenance Organization	Person Approved by Canadian Department of Transport
	FAA Designee	X Repair Station	Inspection Authorization	Other (Specify)

Certificate or Designation No.	Signature/Date of Authorized Individual <i>HCD</i> 11/11/10			
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NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

USA N959JT

11/11/08

Nationality and Registration Mark

Date

CESSNA T206H- T20608983- N959JT

INSTALLED SYSTEMS:

- Installed Atlantic Aero Installation of Cabin Skylights IAW STC SA03123AT-D. Installation completed IAW Atlantic Aero Master Drawing List 23793001 Rev D dated 6/4/08.
- Installed Atlantic Aero Installation of a Rear Camera Window IAW STC SA03122AT-D. Installation completed IAW Atlantic Aero Master Drawing List 23993001 Rev A dated 6/4/08.

CONTINUED AIRWORTHINESS INSTRUCTIONS:

- Reference Atlantic Aero Document No. 23793010 Rev.-A dated 6/8/05 for Instructions for Continued Airworthiness Installation of Skylights in the Cessna 206H.
- Reference Atlantic Aero Document No. 23993010 Rev. A dated 6/25/08 for Instructions for Continued Airworthiness. Installation of a rear Camera Window in the Cessna 206H and T206H.

WEIGHT & BALANCE and EQUIPMENT LIST: Revised Aircraft Weight & Balance and Equipment List. See Aircraft Weight & Balance records for details.

The above installation meets the requirements for static loading in accordance with A.C.43.13-2B Chapter 1 par. 106 through 114. No changes were noted to the compass system. Further details are on file at C.R.S. # YN8R621Y under W.O. # AVI 10096.

>>>>>>>>>>END<<<<<<<<<

Additional Sheets Are Attached



U.S. Department of
Transportation
Federal Aviation
Administration

**MAJOR REPAIR AND ALTERATION
(Airframe, Powerplant, Propeller, or Appliance)**

Form Approved
OMB No. 2120-0020
11/30/2007

Electronic Tracking Number

For FAA Use Only

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act 1958).

1. Aircraft	Nationality and Registration Mark USA N959JT	Serial No. T20608983
	Make CESSNA	Model T206H
2. Owner	Name (As shown on registration certificate) CESSNA AIRCRAFT COMPANY	Address (As shown on registration certificate) Address ATTN: DEPT 093 3 CESSNA BLVD City WICHITA State KANSAS Zip 67215-1400 Country USA

3. For FAA Use Only

4. Type		5. Unit Identification			
Repair	Alteration	Unit	Make	Model	Serial Number
<input type="checkbox"/>	<input checked="" type="checkbox"/>	AIRFRAME	_____	(As described in Item 1 above)	_____
<input type="checkbox"/>	<input type="checkbox"/>	POWERPLANT	_____	_____	_____
<input type="checkbox"/>	<input type="checkbox"/>	PROPELLER	_____	_____	_____
<input type="checkbox"/>	<input type="checkbox"/>	APPLIANCE	Type	_____	_____
<input type="checkbox"/>	<input type="checkbox"/>		Manufacturer		

6. Conformity Statement					
A. Agency's Name and Address			B. Kind of Agency		
Name YINGLING AVIATION Address 2010 AIRPORT ROAD City WICHITA State KS Zip 67277 Country USA			<input type="checkbox"/> U.S. Certificated Mechanic	<input type="checkbox"/> Manufacturer	
			<input type="checkbox"/> Foreign Certificated Mechanic	C. Certificate No.	
			<input checked="" type="checkbox"/> Certificated Repair Station	YN8R621Y	
			<input type="checkbox"/> Certificated Maintenance Organization	RADIO CLASS 1,2,3	

D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.					
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Extended range fuel per 14 CFR Part 43 App. B	<input type="checkbox"/>	Signature/Date of Authorized Individual <i>George L. Maske 11/11/10</i>			
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7. Approval for Return to Service					
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Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is <input checked="" type="checkbox"/> APPROVED <input type="checkbox"/> REJECTED					
---	--	--	--	--	--

BY	FAA Fit Standards Inspector	Manufacturer	Maintenance Organization	Person Approved by Canadian Department of Transport
	FAA Designee	X Repair Station	Inspection Authorization	Other (Specify)
Certificate or Designation No. YN8R621Y		Signature/Date of Authorized Individual <i>H. Collier 11/11/10</i>		

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

USA N959JT

14/11/10

Nationality and Registration Mark

Date

CESSNA T206H – T20608983– N959JT

Provisions: Installed antenna doublers under the fuselage for future install of a Directional Finder Antenna.

2 ea provisions at FSS 31.5

2 ea provisions at FSS 44.0

Cut 1 ea opening provision for DF antenna coax. Located on pilot side under the fuselage at FSS 48.0. Fabricated and installed cover plate.

Reference:

AC 43.13-1B

AC 43.13-2B Chpt 3

The above installation meets the requirements for static loading in accordance with A.C.43.13-2B Chapter 1 par. 106 through 114. No changes were noted to the compass system. Further details are on file at C.R.S. # YN8R621Y under W.O. # AVI-10096.

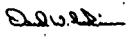
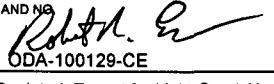
>>>>>>>END<<<<<<<<

Additional Sheets Are Attached

FAA FORM 8130-6, APPLICATION FOR U.S. AIRWORTHINESS CERTIFICATE

Form Approved O.M.B. No. 2120-0018

12/31/2010

APPLICATION FOR U. S. AIRWORTHINESS CERTIFICATE				INSTRUCTIONS - Print or type. Do not write in shaded areas; these are for FAA use only. Submit original only to an authorized FAA Representative. If additional space is required, use attachment. For special flight permits complete Sections II, VI and VII as applicable.												
I. AIRCRAFT DESCRIPTION	1. REGISTRATION MARK N959JT		2. AIRCRAFT BUILDER'S NAME (Make) Cessna Aircraft Company		3. AIRCRAFT MODEL DESIGNATION T206H		4. YR MFR 2010	FAA CODING								
	5. AIRCRAFT SERIAL NO. T20608983		6. ENGINE BUILDER'S NAME (Make) Lycoming Engines		7. ENGINE MODEL DESIGNATION TIO-540-AJ1A											
	8. NUMBER OF ENGINES One		9. PROPELLER BUILDER'S NAME (Make) McCauley Propeller Systems		10. PROPELLER MODEL DESIGNATION B3D36C432/80VSA-1		11. AIRCRAFT IS (Check if applicable)		IMPORT							
APPLICATION IS HEREBY MADE FOR: (Check applicable items)																
II. CERTIFICATION REQUESTED	A	<input checked="" type="checkbox"/>	STANDARD AIRWORTHINESS CERTIFICATE (Indicate category)			<input type="checkbox"/>	NORMAL	UTILITY	ACROBATIC	TRANSPORT	COMMUTER	BALLOON	OTHER			
	B	<input type="checkbox"/> SPECIAL AIRWORTHINESS CERTIFICATE (Check appropriate items)														
	7		PRIMARY													
	9		LIGHT-SPORT (Indicate Class)		AIRPLANE		POWER-PARACHUTE		WEIGHT-SHIFT CONTROL		GLIDER		LIGHTER THAN AIR			
	2		LIMITED													
	5		PROVISIONAL (Indicate class)		1	Class I										
	3		RESTRICTED (Indicate operation(s) to be conducted)		2	Class II										
	4		EXPERIMENTAL (Indicate operation(s) to be conducted)		1	AGRICULTURE AND PEST CONTROL				2	AERIAL SURVEY		3	AERIAL ADVERTISING		
	6		SPECIAL FLIGHT PERMIT (Indicate operation(s) to be conducted, then complete Section VI or VII as applicable on reverse side)		4	FOREST (Wildlife conservation)				5	PATROLLING		6	WEATHER CONTROL		
	8		TO SHOW COMPLIANCE WITH THE CFR		0	OTHER (Specify)										
8		OPERATING LIGHT-SPORT		1	RESEARCH AND DEVELOPMENT				2	AMATEUR BUILT		3	EXHIBITION			
8		OPERATING LIGHT-SPORT		4	AIR RACING				5	CREW TRAINING		6	MARKET SURVEY			
8		OPERATING LIGHT-SPORT		0	TO SHOW COMPLIANCE WITH THE CFR								7	OPERATING (Primary Category) KIT BUILT AIRCRAFT		
8		OPERATING LIGHT-SPORT		8A	Existing Aircraft without an airworthiness certificate & do not meet § 103.1											
8		OPERATING LIGHT-SPORT		8B	Operating Light-Sport Kit-Built											
8		OPERATING LIGHT-SPORT		8C	Operating light-sport previously issued special light-sport category airworthiness certificate under § 21.90											
C		MULTIPLE AIRWORTHINESS CERTIFICATE (check ABOVE: "Restricted Operation" and "Standard" or "Limited" as applicable)														
III. OWNER'S CERTIFICATION	A. REGISTERED OWNER (As shown on certificate of aircraft registration)				IF DEALER, CHECK HERE <input type="checkbox"/>				x							
	NAME Cessna Aircraft Company				ADDRESS 1 Cessna Blvd., PO Box 1996, Independence, KS 67301											
	B. AIRCRAFT CERTIFICATION BASIS (Check applicable blocks and complete items as indicated)															
	<input checked="" type="checkbox"/> AIRCRAFT SPECIFICATION OR TYPE CERTIFICATE DATA SHEET (Give No. and Revision No.) A4CE – Revision 47				<input checked="" type="checkbox"/> AIRWORTHINESS DIRECTIVES (Check if all applicable AD's are complied with and give the number of the last AD SUPPLEMENT available in the biweekly series as of the date of application) 2010-20 Sep 27, 2010											
	C. AIRCRAFT LISTING (Give page number(s)) N/A				SUPPLEMENTAL TYPE CERTIFICATE (List number of each STC incorporated) SA01700LA											
	D. CERTIFICATION - I hereby certify that I am the registered owner (or his agent) of the aircraft described above, that the aircraft is registered with the Federal Aviation Administration in accordance with Title 49 of the United States Code 44101, et seq., and applicable Federal Aviation Regulations, and that the aircraft has been inspected and is airworthy and eligible for the airworthiness certificate requested.															
	DATE OF APPLICATION Oct 1, 2010			NAME AND TITLE (Print or type) David W. LaPierre Quality Manager, Independence				Signature 								
	E. THE AIRCRAFT DESCRIBED ABOVE HAS BEEN INSPECTED AND FOUND AIRWORTHY BY (Complete the section only if 14 CFR part 21.183(d) applies).															
	2		14 CFR PART 121 CERTIFICATE HOLDER (Give Certificate No.)		3	CERTIFICATED MECHANIC (Give Certificate No.)		6	CERTIFICATED REPAIR STATION (Give Certificate No.)							
	5		AIRCRAFT MANUFACTURER (Give name or firm)													
F. DATE			TITLE			SIGNATURE										
V. FAA REPRESENTATIVE CERTIFICATION	(Check ALL applicable block items A and B) A. I find that the aircraft described in Section I or VII meets requirements for				<input checked="" type="checkbox"/>	THE CERTIFICATE REQUESTED										
	B. Inspection for a special flight permit under Section VII was conducted by:				<input type="checkbox"/>	FAA INSPECTOR				<input checked="" type="checkbox"/>	FAA DESIGNEE					
					CERTIFICATE HOLDER UNDER				14 CFR part 65	14 CFR part 121 or 135	14 CFR part 145					
	DATE Oct 1, 2010		MIDO/FSDO Office CE-43	4	DESIGNEE'S SIGNATURE AND NO.  Robert R. Evans ODA-100129-CE				1	FAA INSPECTOR'S SIGNATURE 						

VI PRODUCTION FLIGHT TESTING	A MANUFACTURER			
	NAME		ADDRESS	
	B. PRODUCTION BASIS (Check applicable item)			
		PRODUCTION CERTIFICATE (Give production certificate number) →		
		TYPE CERTIFICATE ONLY		
		APPROVED PRODUCTION INSPECTION SYSTEM		
	C. GIVE QUANTITY OF CERTIFICATES REQUIRED FOR OPERATING NEEDS			
	DATE OF APPLICATION	NAME AND TITLE (Print or type)		SIGNATURE
A. DESCRIPTION OF AIRCRAFT				
REGISTERED OWNER		ADDRESS		
BUILDER (Make)		MODEL		
SERIAL NUMBER		REGISTRATION MARK		
B. DESCRIPTION OF FLIGHT		CUSTOMER DEMONSTRATION FLIGHTS <input type="checkbox"/> (Check if applicable)		
FROM		TO		
VIA		DEPARTURE DATE	DURATION	
c. CREW REQUIRED TO OPERATE THE AIRCRAFT AND ITS EQUIPMENT				
	PILOT	CO-PILOT	FLIGHT ENGINEER	OTHER (Specify)
D. THE AIRCRAFT DOES NOT MEET THE APPLICABLE AIRWORTHINESS REQUIREMENTS AS FOLLOWS:				
E. THE FOLLOWING RESTRICTIONS ARE CONSIDERED NECESSARY FOR SAFE OPERATION: (Use attachment if necessary)				
F- CERTIFICATION - I hereby certify that I am the registered owner (or his agent) of the aircraft described above; that the aircraft is registered with the Federal Aviation Administration in accordance with Title 49 of the United States Code 44101 <u>et seq.</u> and applicable Federal Aviation Regulations; and that the aircraft has been inspected and is safe for the flight described.				
DATE	NAME AND TITLE (Print or type)		SIGNATURE	
VII AIRWORTHINESS DOCUMENTATION/FAA DESIGNEE USE only	X A. Operating Limitations and Markings in Compliance with 14 CFR section 91.9, as Applicable		G- Statement of Conformity, FAA Form 8130-9 (Attach when required)	
	B. Current Operating Limitations Attached		H. Foreign Airworthiness Certification for Import Aircraft (Attach when required)	
	C. Data, Drawings, Photographs, etc. (Attach when required)		I. Previous Airworthiness Certificate issued in Accordance with 14 CFR Section _____ CAR _____ (Original attached)	
	D. Current Weight and Balance Information Available in Aircraft			
	E. Major Repair and Alteration, FAA Form 337 (Attach when required)		J. Current Airworthiness Certificate Issued in Accordance with 14 CFR Section <u>21.183 (a)</u>	
	F. This inspection Recorded in Aircraft Records		K. Light-Sport Aircraft Statement of Compliance, FAA Form 8130-15 (Attach when required).	

UNITED STATES OF AMERICA
DEPARTMENT OF TRANSPORTATION-FEDERAL AVIATION ADMINISTRATION
STANDARD AIRWORTHINESS CERTIFICATE

1 NATIONALITY AND REGISTRATION MARKS	2 MANUFACTURER AND MODEL	3 AIRCRAFT SERIAL NUMBER	4 CATEGORY
N959JT	Cessna Aircraft Company T206H	T20608983	Normal

5 AUTHORITY AND BASIS FOR ISSUANCE

This airworthiness certificate is issued pursuant to the Federal Aviation Act of 1958 and certifies that, as of the date of issuance, the aircraft to which issued has been inspected and found to conform to the type certificate therefor, to be in condition for safe operation, and has been shown to meet the requirements of the applicable comprehensive and detailed airworthiness code as provided by Annex 8 to the Convention on International Civil Aviation, except as noted herein.

Exceptions:

None

DUPLICATE

6 TERMS AND CONDITIONS

Unless sooner surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator, this airworthiness certificate is effective as long as the maintenance, preventative maintenance, and alterations are performed in accordance with Parts 21, 43, and 91 of the Federal Aviation Regulations, as appropriate, and the aircraft is registered in the United States.

DATE OF ISSUANCE	FAA REPRESENTATIVE	DESIGNATION NUMBER
Oct 1, 2010	Robert R. Evans 	ODA-100129-CE

Any alteration, reproduction, or misuse of this certificate may be punishable by a fine not exceeding \$1,000 or imprisonment not exceeding 3 years or both.
THIS CERTIFICATE MUST BE DISPLAYED IN THE AIRCRAFT IN ACCORDANCE WITH APPLICABLE FEDERAL AVIATION REGULATIONS.

