AIRFRAME ANALYSIS

Nathan Pettit

1   Introduction

This report covers research done on airframes using the VortexLattice.jl library, specifically how aspect ratio and efficiency are related, how tail volume ratios affect stability derivatives, and how angle of attack affects the lift coefficient. A wiki with unfamiliar terms and their corresponding equations and figures can be found at the end of this report.

2   Methods

The results of this research came from evaluating airframes using VortexLattice.jl, as well as writing new functions to help in evaluating the airframes. There were 4 functions that were used:

1. aoaEffect() - this evaluated airframes and calculated their coefficient of lift for varying angles of attack

2. wingEfficiency() - this calculated the efficiency of airframes with varying aspect ratios; it did this by calculating the lift and drag coefficients for every aspect ratio

3. stabilityDerivatives() - this calculated the stability derivatives for airframes with different horizontal and vertical tail ratios

4. vortexLattice() - this was the main function that performed all the necessary VortexLattice calculations for the other functions

3   Results and Discussion

3.1   Aspect Ratio vs. Efficiency

In order to see the relationship between aspect ratio and the efficiency, the wingEfficiency() function was used. When it was run, it calculated the inviscid span efficiency (see equation 2) for airframes with aspect ratios (see equation 1) ranging from 3 to 15. it produced a plot of this shown in figure 1.
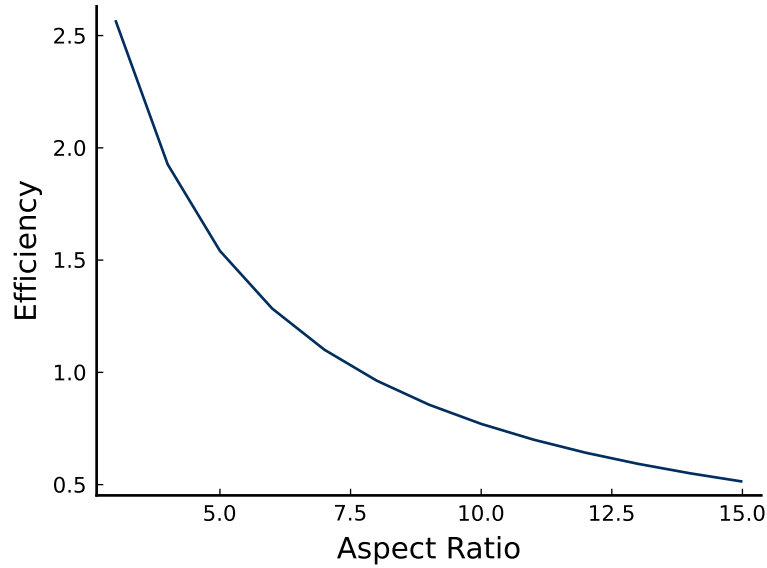
Fig. 1: *This diagram shows the relationship between aspect ratio and inviscid span efficiency.*

In reality, there is actually no physical relationship between an airframe's aspect ratio and its efficiency, but figure 1 does mathematically plot it correctly. However, that plot is simply the definition of inviscid span efficiency.

3.2   Tail Volume Ratios vs Stability Derivatives

Another characteristic of these airframes that was observed was their stability derivatives. Stability derivatives were calculated for airframes with different tail volume ratios, both vertical and horizontal. The definitions of those ratios are equations 3 and 4. The stability derivatives that are affected by a change in vertical tail volume ratio are $C_{\ell b}$ and C_{nb}. An airframe is stable if $C_{\ell b}$ is less than 0 and C_{nb} is greater than 0.

In figure 2 , we can see that the airframe is stable for vertical tail volume ratios greater than 0.003, but it becomes unstable when the ratio is less than 0.003, at around 0.002. The stability derivatives that are affected by a change in horizontal tail volume ratio are C_{La} and C_{ma}. An airframe is stable if C_{La} is greater than 0 and C_{ma} is less than 0. Figure 3 shows that the airframe is stable for horizontal tail volume ratios greater than 0.05, but becomes unstable when the ratio is less than 0.05, at around 0.025.

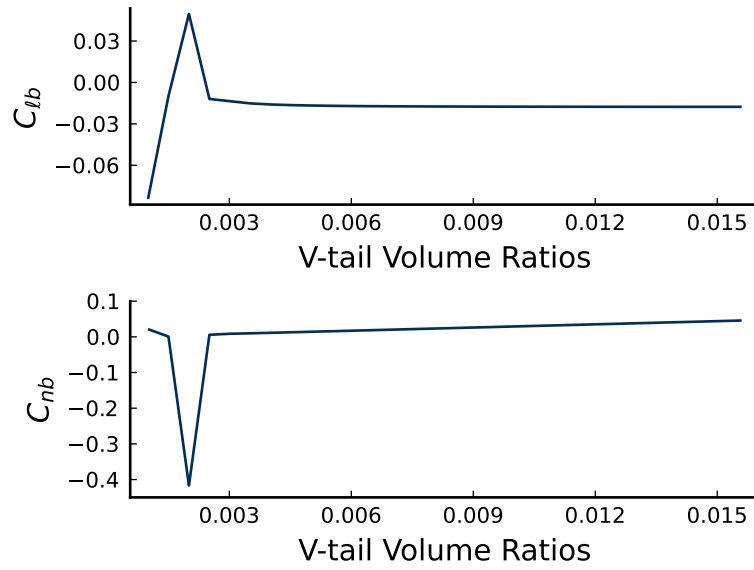These ratios are arbitrary and will be different among models, but they are

Fig. 2: *This diagram shows the relationship between vertical tail volume ratio, $C_{\ell b}$, and C_{nb}.*

still extremely important.

3.3 Angle of attack vs lift coefficient

In previous research about airfoils, I found that as angle of attack increases, the lift coefficient also increases, up to a certain point. At that point, the lift coefficient sharply drops, and this is where stall takes place. In that previous research, Xfoil.jl was used to calculate the lift coefficient for varying angles of attack. A relationship between angle of attack and lift was also tried to be found here, but instead of using Xfoil.jl, VortexLattice.jl was used. A plot was produced showing the relationship between angle of attack and lift (see figure 4).
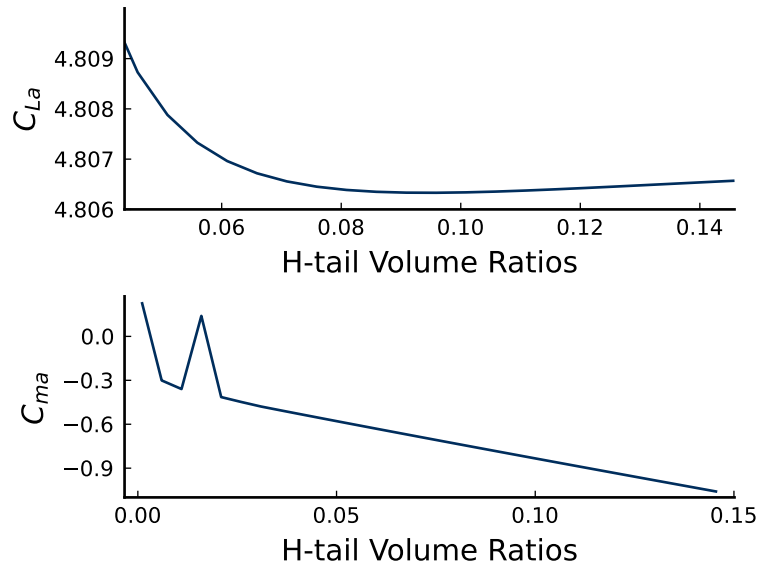
Fig. 3: *This diagram shows the relationship between horizontal tail volume ratio, C_{La}, and C_{ma}.*
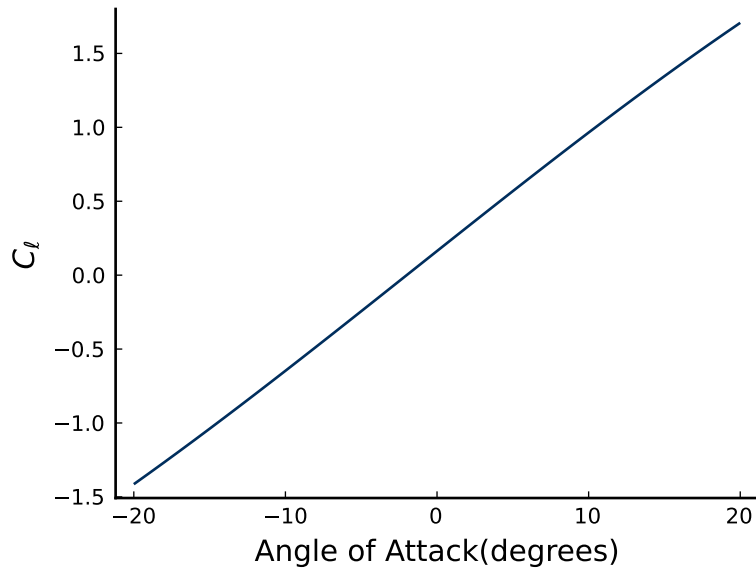

Fig. 4: *This diagram shows the relationship between angle of attack and lift for an airframe.*

In comparison to the plot created using Xfoil.jl, figure 4 looks very similar

between angles of attack of -10 and 10. However, past those angles of attack is where it differs. In the Xfoil plot, it shows very clearly where stall would take place, but in this VortexLattice plot, the plot continues linearly and never drops off, never indicating stall. This is why the Vortex Lattice method is not always a great choice for investigating airframes. It is only accurate when the angle of attack and sideslip angle are small, because it uses small angle approximation in its calculations. This is why in figure 4 it looks very linear and doesn't indicate where stall would take place, where as the Xfoil plot does.

Appendix

- Sweep: the angle at which the wing is translated backwards (or occasionally forwards) relative to the root chord of the wing (see figure 5)
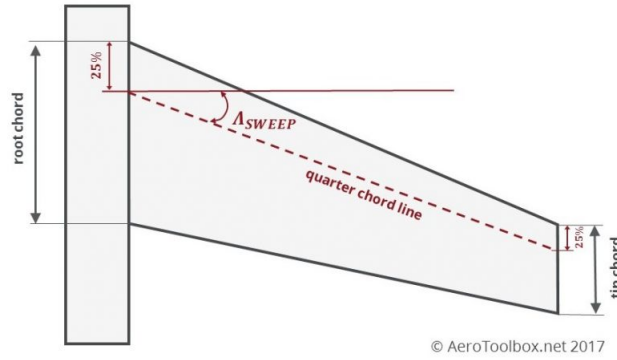

Fig. 5: *This diagram shows what sweep is on an airframe.*

- Aspect Ratio: aspect ratio of a wing is the ratio of its span to its mean chord. It is equal to the square of the wingspan divided by the wing area. Thus, a long, narrow wing has a high aspect ratio, whereas a short, wide wing has a low aspect ratio (see equation 1)

$$AR = \frac{b}{c} = \frac{b^2}{S_{ref}} \qquad (1)$$

- Wake Vortex/Tip Vortex: a lifting wing has higher pressure on the bottom surface as compared to the top, and so fluid will circulate around the tips (see figure 6)

- Downwash: can be viewed more fundamentally as a consequence of Newton's third law. If a body is producing lift, colloquially we might say that means that the air is pushing the body up. Thus, by Newton's third law, the body must be pushing the air downward. Thus, any three-dimensional
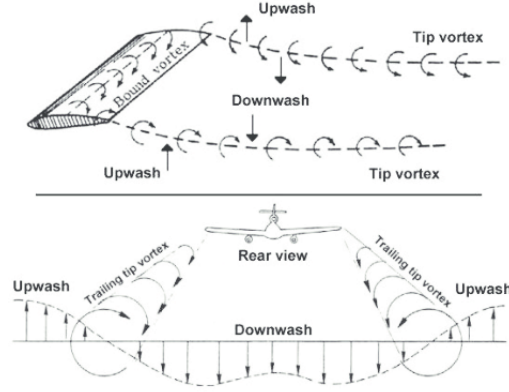
Fig. 6: *This diagram shows how downwash and wake vortexes are formed.*

lifting body will leave behind a wake of downward moving air (see figure 6)

- Induced Drag/Vortex Drag: major consequence of this downwash, is that energy is left behind in the wake. Or in other words, the wing produces drag

- Inviscid Span Efficiency: the inviscid span efficiency could be considered a measure of how close the lift distribution is to elliptic. All planar distributions will have $e_{inv} <= 1$. A nonplanar lift distribution (e.g., adding winglets) can increase the inviscid span efficiency above 1 (see equation 2)

$$e_{inv} = \frac{C_L^2}{\pi AR C_D} \tag{2}$$

- Vortex Filament: an arbitrary vortex line segment

- Vortex Lattice Method: an extension of thin airfoil theory into three dimensions

- Tail Volume Ratio: relates the shape of the tail to the wing

    Vertical Tail Volume Ratio:

$$V_v = \frac{l_v S_v}{Sb} \tag{3}$$

    Horizontal Tail Volume Ratio:

$$V_h = \frac{l_t S_t}{SC_{ma}} \tag{4}$$

- Stability Derivatives: measures of how particular forces and moments on an aircraft change as other parameters related to stability change