

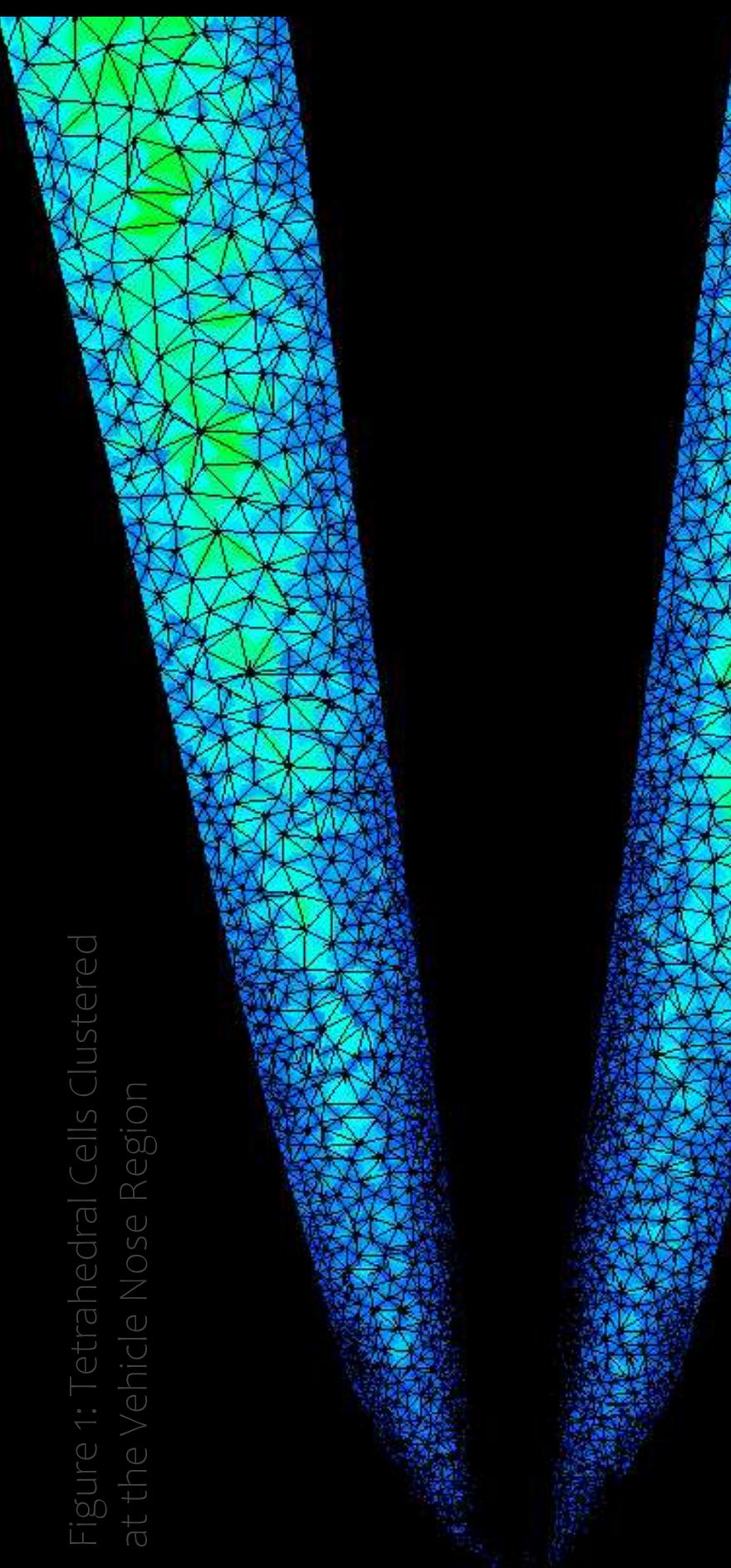
Hypersonic Flow Modeling with Unstructured Grids

Rapid and accurate grid generation for re-entry vehicles

Priscilla Pak* & Geoffrey Andrew†

*Senior, Boston University
†Technical Staff, MIT Lincoln Laboratory

Figure 1: Tetrahedral Cells Clustered at the Vehicle Nose Region



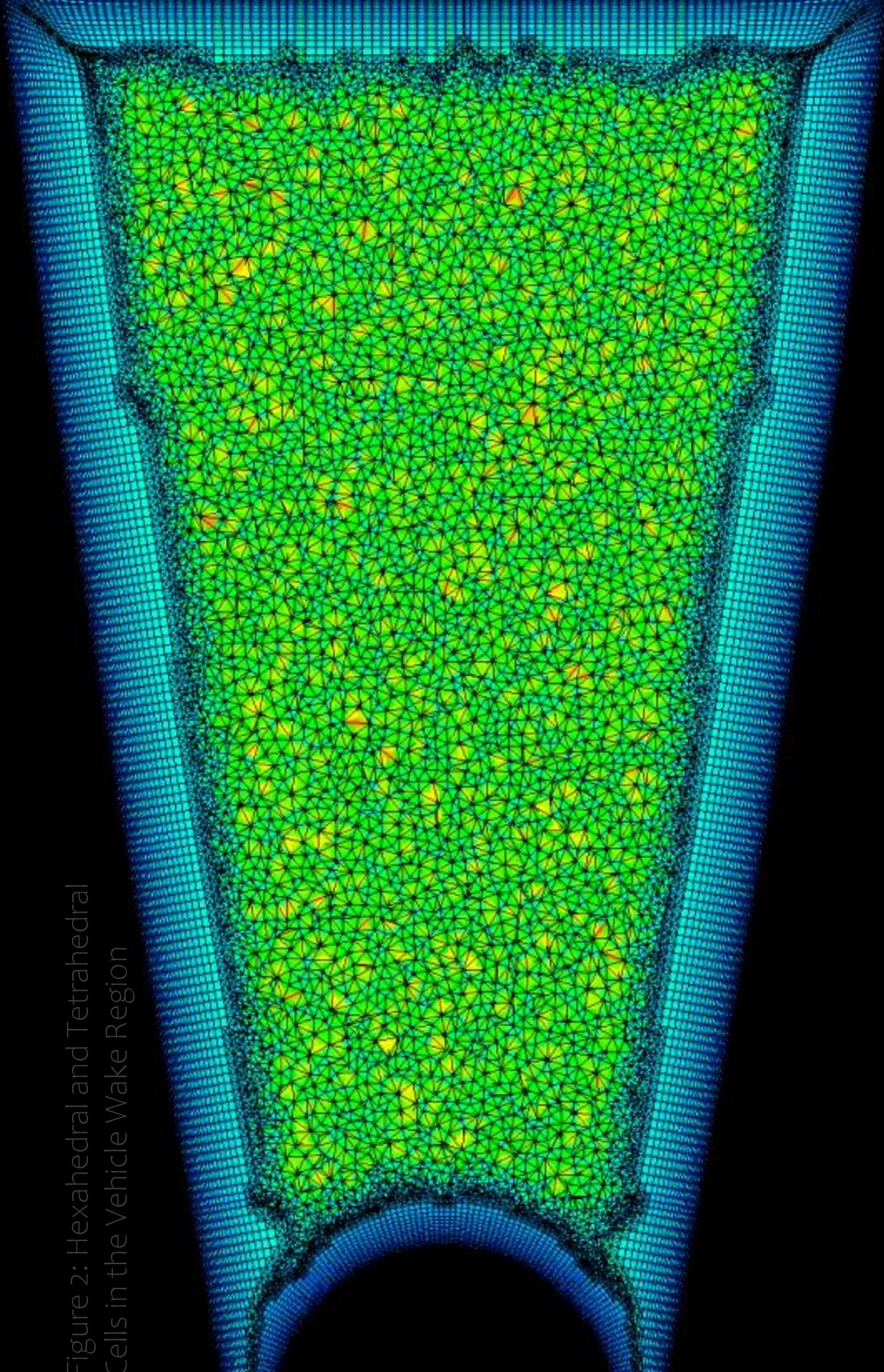
Problem

The high temperatures and pressures in hypersonic flow fields are difficult and expensive to measure during flight testing.

CFD models of hypersonic flows may produce a more detailed, less expensive picture of flight phenomena. However, accurate CFD models require careful setup, including satisfying stringent meshing requirements.

Unstructured meshes pose a viable alternative to structured meshes, which may take longer to adapt to a geometry.

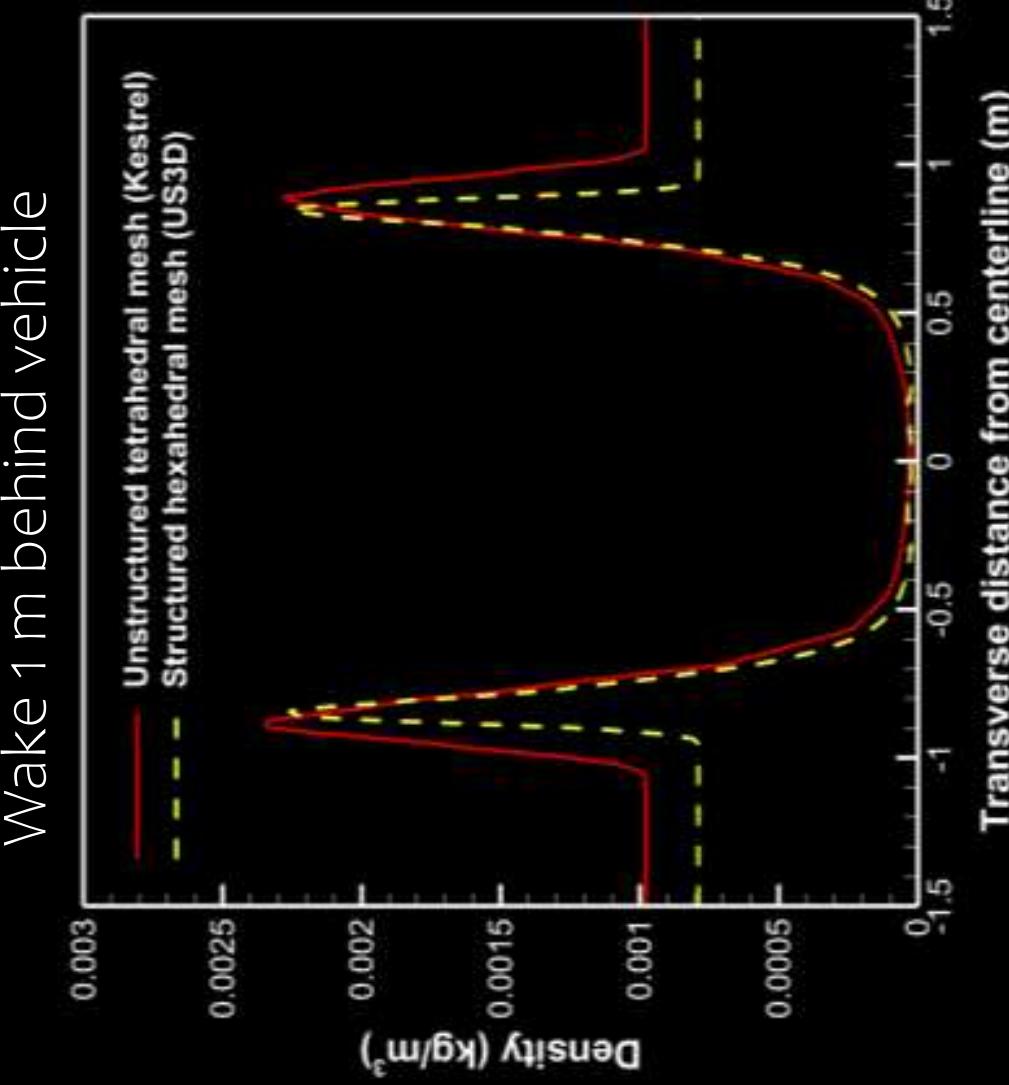
Figure 2: Hexahedral and Tetrahedral Cells in the Vehicle Wake Region



Method

Hexahedral and prismatic cells are extruded from a vehicle body to resolve the boundary layer, while the bow shock and wake are resolved in the farfield using both tetrahedral and hexahedral cells.

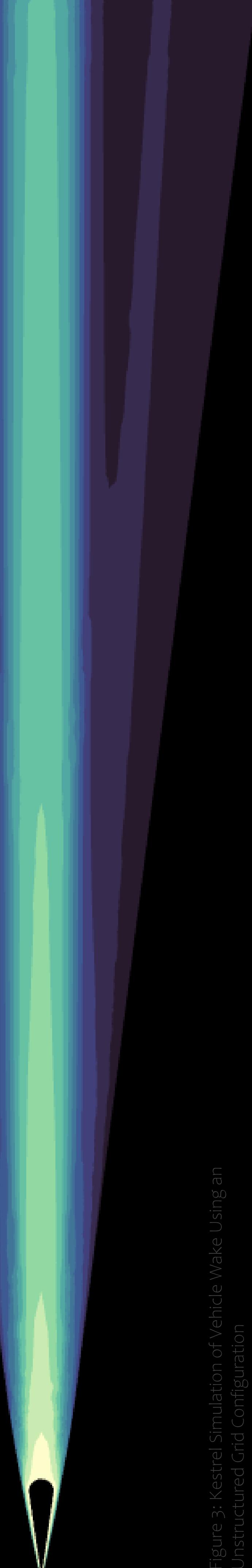
Results



Tetrahedral cells proved to have acceptable accuracy with appropriate clustering but prismatic and hexahedral cells demonstrated superior resolution of the boundary layer with less mesh generation time and storage.

This approach is demonstrated for a sharp cone-sphere at Mach 21.

Figure 3: Kestrel Simulation of Vehicle Wake Using an Unstructured Grid Configuration



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