

# Near-field Decomposition of an Excited Mach 0.9 Jet - Complementary Experimental and Computational Analysis

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## I. Introduction

Acoustic radiation has been a significant problem for commercial airports and aircraft carriers since the advent of turbojet engines in aeronautical applications. Commercial aviation must contend with more restrictive curfews, higher surcharges and flight path restrictions in order to appease the urban and residential areas near the airports which are perturbed by the jet noise. On aircraft carriers, the close proximity of ground crew personnel to tactical jet engines during take-off and landing creates a high potential for hearing loss. To address these issues, noise reduction concepts, most notably passive control techniques such as tabs, chevrons or lobed mixers, have been implemented in commercial vehicles, and military applications are currently being investigated. However, these modifications have a non-trivial penalty to the engine's performance, either in terms of increased weight or decreased thrust. Therefore, active control techniques are currently being investigated in order to mitigate the losses seen from passive control. In order to effectively control the production of noise, the mechanisms that produce aeroacoustic pollution must be well understood.

Lighthill<sup>1</sup> was the first to show that the governing equations for fluid dynamics could be rearranged into an inhomogenous wave equation. The source term, later dubbed Lighthill's stress tensor, comprises Reynolds stress, shear stress, and density fluctuations terms. Aside from the assumption of a constant sound speed, this rearrangement is exact. Therefore, complete knowledge of the source term should yield an exact solution for the acoustic far-field. In practical applications however, the full source term is unknown and certain simplifications are required. Upon the identification of coherent structures in turbulent jet shear layers,<sup>2-5</sup> source term models based on coherent eddies have frequently been employed, to varying success. Recognizing that the acoustic far-field spectra of supersonic and subsonic jets can be represented as two distinct universal similarity spectra, independent of jet Mach number or temperature ratio, Tam *et al.*<sup>6</sup> proposed a two-component source mechanism for jet mixing noise. Large-scale coherent structures (alternatively represented as instability waves) being primarily responsible for the aft angle radiation, produce spectra with a distinct amplitude peak. Fine-scale turbulence, on the other hand, produces a less coherent, more broadband acoustic field and is the dominant source of acoustic radiation at sideline angles. Experiments utilizing direct correlations between density and velocity fluctuations in the shear layers of high-speed jets and the acoustic far-field have supported this two-component source model.<sup>7,8</sup>

The mechanism by which the acoustic sources produce the acoustic radiation is not well understood in the case of the subsonically-convective jet. The highly directional and highly coherent radiation produced by the large-scale structures is congruent with Mach-wave radiation, and it was initially proposed that a similar mechanism was indeed the source mechanism in both supersonic and subsonic jets.<sup>9</sup> These axially extended waveforms have been identified as having wavepacket characteristics,<sup>10</sup> which has led to the frequency-domain description of the large-scale structures. In the subsonic jet, modulation of the waveform as it is advecting through space is necessary to produce radiation to the far-field. Spatial modulation of the

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wavepacket was shown to produce the superdirective character of far-field acoustic radiation observed in subsonic jets.<sup>11</sup> Similarly, temporal modulation of the wavepacket, in terms of amplitude and envelope, has been shown to improve agreement between analytic models and the numerical results in terms of directivity and amplitude.<sup>12,13</sup>

Understanding the exact spatiotemporal evolution of the large-scale structures is important to predicting and ultimately controlling their radiation production. However, in a highly turbulent jet identifying the aeroacoustic source terms is not trivial due to the dissimilar range of scales, fluctuation intensities, and structure lifetimes. In the irrotational near-field of the jet, strong evanescent waves (pseudo-sound) associated with the advection of large-scale coherent structures in the shear layer coincide with the resultant acoustic field. Therefore, in order to identify pure acoustic waves traveling to the far-field and their corresponding source events, a decomposition of the irrotational near-field into its constitutive components is required. By identification and prediction of coherence nulls between microphones in the near-field, Coiffet *et al.*<sup>14</sup> showed that the full irrotational near-field consists primarily as a linear superposition of its hydrodynamic and acoustic components. This led subsequent researchers to propose linear filters to extract the individual components from the near-field pressure, with varying degrees of success. Tinney and Jordan<sup>15</sup> used a Fourier-based wavenumber-frequency filter method in a cold, subsonic jet to separate the the near-field pressure into supersonically- and subsonically-conveying waves (and hence, the hydrodynamic and acoustic components). The method of Kuo *et al.*<sup>16</sup> dispensed with explicit concerns with the phase velocity of the pressure components and instead decomposed the field using Empirical Mode Decomposition and the critical frequency, as defined by Arndt *et al.*,<sup>2</sup> which demarcates the energy dominance of the acoustic and hydrodynamic components in the near-field spectra. In the work of Crawley and Samimy,<sup>17</sup> a new decomposition method was proposed in which a spatio-temporal continuous wavelet transform is used to filter the irrotational pressure acquired experimentally in an unheated Mach 0.9 jet. They compared this new method against those of Tinney and Jordan<sup>15</sup> and Kuo *et al.*<sup>16</sup> and found the new method to be superior for decomposing the broadband pressure field, as well as identifying and reconstructing strongly-energetic localized bursts of energy.

Post-processing experimental and computational data using reduced order models is beneficial to looking at the jet noise mechanisms in a new way. For instance, Jordan *et al.*<sup>18</sup> developed a method called Most Observable Decomposition (MOD) in order to better capture the modes that generate noise. They found with this technique that only 24 MOD modes were required to capture 90% of the sound energy of a Mach 0.9 turbulent jet. Of this sound energy captured, 48% is from the break up of the coherent structures after the potential core and 12% is from the convection of coherent structures. Another approach to understanding the noise sources of jets has been Proper Orthogonal Decomposition (POD) of the flow variables in the Fourier domain. This method maintains the frequency spectra and ranks modes by energy. With this method, Arndt<sup>2</sup> examined the pressure signal outside a mixing layer and found that the phase velocity was constant indicating that the jet is non-dispersive. Others have found and confirmed that the  $m = 0$  mode is dominate in the low wavenumbers and contributes the most to large scale structure noise.<sup>19,20</sup> Using the DNS of a Mach 0.9 jet, Freund and Colonius<sup>21</sup> discovered that the modes come in pairs and resemble a wavepacket structure. This method has also been applied before to study the control of jets. Moreno *et al.*<sup>22</sup> computed the POD of far-field microphones of no-control and microjet controlled supersonic jets and observed a 67% decrease in energy for the first mode while using microjet control. Other methods are also combined with POD (i.e. linearized stochastic estimation (LSE) and high-order stochastic estimation (HOSE)) in order to expand the available findings. Baars and Tinney<sup>23</sup> performed a POD based spectral High-Order Stochastic Estimation (HOSE) on experimental data simultaneously obtained in the near and far-fields and determined that the hydrodynamic waveforms in the near-field are linearly related to the far-field signal.

In this work, we look to combine the analysis tools of wavelet-based hydrodynamic and acoustic decomposition with that of POD in the Fourier domain in order to study a subsonic jet. The jet has been controlled with plasma actuators using the axisymmetric mode ( $m=0$ ) in order to seed the growth of large-scale structures in the jet shear layer. Concurrently, numerical simulations have been conducted using LES, which match both the jet and the actuation conditions of the experiments. Through the use of various postprocessing techniques, the near-field hydrodynamic and far-field acoustic response of the jet to excitation has been analyzed over a range of structure frequencies. Preliminary work has been completed in order to decompose the irrotational near-field of the jet into its constitutive hydrodynamic and acoustic components, and to begin identification of the dominant acoustic source region and acoustic emission events. The experimental and numerical databases will complement one another, as the relevant large-scale structure dynamics which

lead to acoustic emission are identified, and a simplified source model is obtained.

## II. Experimental Setup

All experiments were conducted at the GDTL within the Aerospace Research Center at the Ohio State University. Compressed, dried, and filtered air is supplied to the facility from two cylindrical storage tanks with a total capacity of  $43 \text{ m}^3$  and maximum storage pressure of 16 MPa. The air may be routed through a storage heater (not used in this study), which allows the jet to operate with a stagnation temperature up to  $500^\circ\text{C}$ , before expanding through a nozzle and exhausting horizontally into an anechoic chamber. Opposite the nozzle, a collector accumulates the jet and entrained air from the jet and exhausts it to the outdoors. A schematic of the anechoic chamber can be seen in Figure 1. The dimensions of the chamber are 6.20 m wide by 5.59 m long and 3.36 m tall, with internal wedge-tip to wedge-tip dimensions of 5.14 m by 4.48 m and 2.53 m, respectively. The design of the chamber produces a cutoff frequency of 160 Hz, well below the frequencies of interest. A more detailed description of the GDTL anechoic chamber properties and validation has been given by Hahn.<sup>24</sup>

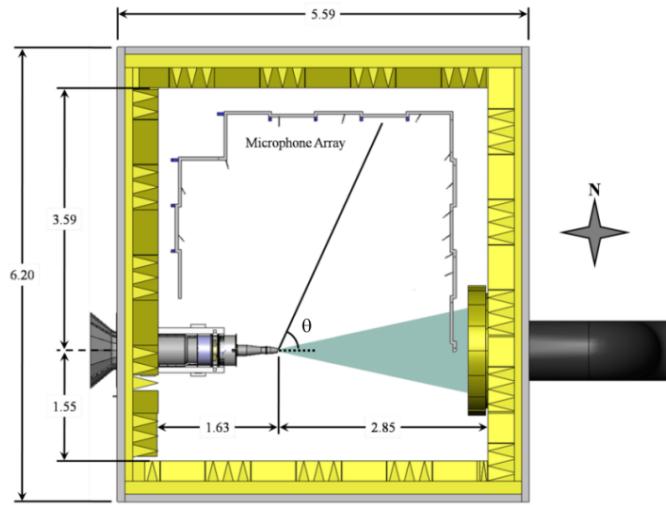


Figure 1: Plan view of the anechoic chamber at the GDTL (dimensions in meters)

For this study a converging, axisymmetric nozzle with exit diameter  $D$  of 25.4 mm (1 in.) was used. The internal contour of the nozzle was designed using a fifth order polynomial. The nozzle utilized a thick-lipped design in order to simplify the mounts for the extension, which housed the eight actuators used in this study. For the experiments reported in this paper, the jet was operated at a Mach number of 0.90, and with a total temperature ratio of unity. The Reynolds number based on the jet exit diameter was ; previous investigations using hot-wire anemometry have indicated that the initial shear layer is turbulent for this operating condition with momentum thickness 0.09 mm and boundary layer thickness 1 mm.<sup>25</sup>

In the present work, the phase-averaging technique used in Sinha *et al.*<sup>26</sup> is employed in order to study the evolution of the seeded perturbations, both spatially and temporally. Points of time within this phase-averaged period (in terms of  $2\pi$ ) will be considered a phase ( $\phi$ ). The TTL pulse sequence, which controls the LAFPAs, is supplied to an Agilent 3320A waveform generator. The rising edge of the TTL pulse triggers a sharp drop in the output voltage of the waveform generator, which then ramps back up to the original voltage over a time interval which is shorter than the minimum forcing period. The output from the waveform generator is acquired simultaneously with the near- and far-field pressure signals using the aforementioned National Instruments hardware and software. As the forcing Strouhal number, azimuthal mode, and ramp signal are well defined, this system enables the identification of the zero phase of actuation and hence, the ability to phase-average the pressure signals over the forcing period. This ensures that the seeded perturbations can be readily identified in the noisy flow, as well as allowing pressure signals, which

were not recorded simultaneously (i.e. different near-field array positions), to be analyzed concurrently.

Analysis of the near-field response of the forced jet is not immediately straightforward due to acoustic contamination from the actuators themselves.<sup>26</sup> LAFPAs operate on a joule heating principle - the breakdown of the air between the electrodes and the ensuing flow of current results in intense heating of the air. This rapid, localized thermal perturbation produces a compression wave, which excites the shear layer. However, this compression wave is still evident as it travels through the near-field. Obviously, this is an undesirable effect, as this actuator self-noise may in some cases obscure the hydrodynamic and acoustic response of the jet. In the present work, the near-field pressure signals have been preprocessed using a continuous-wavelet-based filtering algorithm, which has been specifically designed to remove the actuator self-noise while leaving the response of the jet unaltered. Further details of the filtering algorithm can be found in the references.<sup>27</sup>

## II.A. Localized Arc Filament Plasma Actuators

Each LAFPA consists of a pair of tungsten pin electrodes, which are placed around the nozzle perimeter 1 mm upstream of the nozzle exit. Eight uniformly spaced actuators were used in this study. The center-to-center spacing between electrode pairs for each actuator is 4 mm. The electrodes are housed in a boron nitride extension attached to the end of the nozzle. A groove with dimensions of 1 mm wide and 0.5 mm deep is machined in the boron nitride, into which the electrode tips protrude, to provide a region of low momentum flow in order to stabilize the formation of the plasma arcs. It has been shown that the existence of this groove does not substantially alter the flow field or the control authority of the LAFPAs.<sup>28</sup> A more detailed description of LAFPA characteristics can be found in Utkin *et al.*<sup>29</sup>

The LAFPAs are energized by a multi-channel, high-voltage plasma power generator capable of simultaneously powering up to eight LAFPAs, which was designed and built in-house at the GDTL. In the second-generation power supply, each individual circuit consists of a switchable capacitor in line with a high voltage transformer; the arcing electrodes are connected to the secondary side of the coil. The capacitor is charged by a 100 V DC power supply when the first switch is closed and the second is opened; at the user-specified time the switches flip and it discharges through the coil. The switches are controlled by a 16-channel digital I/O card and National Instruments' Labview software, operated by a dedicated computer. The plasma generator provides independent control of the frequency, duty cycle/pulse width, and phase of each individual actuator (though not the amplitude). The pulse width was held constant at  $7\mu s$ , which was found to be the minimum pulse width at which the actuators consistently arced for all frequencies explored in this study.<sup>28</sup> For reference, this results in a duty cycle of 0.4% at  $St_{DF} = 0.05$  and 2.0% at  $St_{DF} = 0.25$ . The circuit is capable of operating at up to 100 kHz, though presently it is limited to 20 kHz due to thermal concerns.

## II.B. Data Acquisition

Near-field and far-field pressure measurements were acquired simultaneously, using Brüel and Kjær 1/4 inch 4939 microphones. The signal from each microphone is band-pass filtered from 20 Hz to 100 kHz using a Brüel and Kjær Nexus 2690 conditioning amplifier, and recorded using National Instruments PXI-6133 A/D boards and LabView software. The microphones are calibrated using a Brüel and Kjær 114 dB, 1 kHz sine wave generator. The frequency response of the microphones is flat up to roughly 80 kHz, with the protective grid covers removed. Voltage signals are collected at 200 kHz with 81920 data points per block; sub-blocks of 8192 data points were used when calculating short-time power spectral densities, resulting in a frequency resolution of 24.4 Hz. Ten blocks were recorded for each case resulting in four seconds of data, which has been found to be sufficient for convergence of turbulence statistics.

Far-field acoustic pressure was acquired at three polar angles: 30°, 60° and 90°, as measured from the downstream jet axis. The microphones are oriented such that the normal vector from their tips intersects the jet downstream axis at the nozzle exit. The radial distance of the microphones ranges from 101D at 30° to 145D at 60°. The near-field pressure was acquired using a linear array of sixteen microphones located along the meridional plane of the jet; the spacing varied along the array from 1D to 2D (Figure 2a). The linear array was mounted on a linear traverse system at an angle of 8.6° to the jet axis in order to match the spreading angle of the jet shear layer for this Mach number, as determined via PIV during previous studies.<sup>25</sup> The traverse was controlled using LabView and enabled the acquisition of pressure measurements at various radial positions to the jet axis. Initially, the most upstream microphone is positioned at  $x/D = 1$  and  $r/D = 1.20$ , to ensure that the microphone tips are outside the mixing layer and do not affect the flow

field. For subsequent cases, the microphone array was incremented radially outward by  $0.5D$  for a total travel distance of  $7D$ , for a total of 15 microphone locations in the radial direction. A schematic of the microphone locations can be found in Figure 2b.

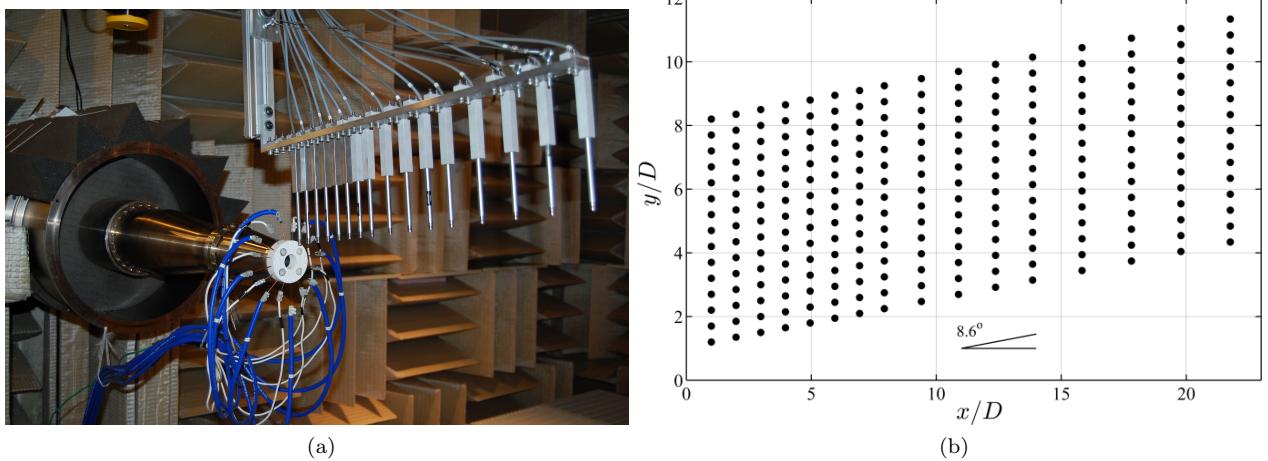


Figure 2: Photograph of anechoic chamber and nozzle, with near-field linear microphone array in foreground (a) and schematic of all near-field microphone locations (b).

### III. Computational Model

The simulations employ the same approach as previously used successfully to simulate a Mach 1.3 jet without and with control.<sup>30,31</sup> The full compressible Navier-Stokes equations are solved in curvilinear coordinates  $(\xi, \eta, \zeta)$  using the strong conservative form.<sup>32,33</sup> The transformed non-dimensional equations in vector notation are given as:

$$\frac{\partial}{\partial \tau} \left( \frac{\vec{U}}{J} \right) + \frac{\partial \hat{F}}{\partial \xi} + \frac{\partial \hat{G}}{\partial \eta} + \frac{\partial \hat{H}}{\partial \zeta} = \frac{1}{Re} \left[ \frac{\partial \hat{F}_v}{\partial \xi} + \frac{\partial \hat{G}_v}{\partial \eta} + \frac{\partial \hat{H}_v}{\partial \zeta} \right] \quad (1)$$

where  $\vec{U} = \{\rho, \rho u, \rho v, \rho w, \rho E\}$  denotes the solution vector and  $J = \partial(\xi, \eta, \zeta, \tau) / \partial(x, y, z, t)$  is the transformation Jacobian. Details of the various terms in Eqn. 1 may be found in Speth and Gaitonde.<sup>34</sup> For the inviscid terms, a third-order upwind biased approach is adopted, together with the Roe scheme<sup>35</sup> for flux evaluation. The limiter required to enforce monotonicity is a crucial component of the method. The van Leer harmonic limiter<sup>36</sup> has proven very successful at reproducing the main features of the unsteadiness in the jet. The viscous terms are discretized with second-order centered differences and time integration is performed by a second-order diagonalized<sup>37</sup> approximately factored method.<sup>38</sup> A sub-iteration strategy is used to minimize errors due to factorization, linearization and explicit boundary condition implementation.

A 65 million point mesh (Fig. 3) is used to simulate the Mach 0.9 jet measured in the experiment (Fig. 2a). The grid has dimensions of 685 points on the  $\xi$  (streamwise) direction, 455 points in the  $\eta$  (radial) direction, and 209 (azimuthal) points in the  $\zeta$  direction. In the radial direction, the mesh is refined in the nozzle region and gradually stretched in the far field. At the exit of the nozzle, the grid maintains a constant axial spacing until after the potential core length; then stretches in the streamwise direction as well. To preserve continuity, the grid has a five point overlap in the  $\zeta$  direction. Characteristic boundary conditions<sup>39</sup> are applied to the upstream (outside the nozzle) and radial boundaries. Non-reflecting conditions are applied to the downstream and far-field boundaries. Stagnation conditions are specified at the first  $\xi$  plane of the nozzle ( $\rho_{inlet} = 2.04 \text{ kg/m}^3$ ,  $U_{inlet} = 22 \text{ m/s}$ ,  $P_{inlet} = 171,427 \text{ Pa}$ ) to achieve perfectly expanded nozzle exit conditions corresponding to  $\rho_{jet} = 1.404 \text{ kg/m}^3$ ,  $U_{jet} = 285.99 \text{ m/s}$ ,  $T_{jet} = 251.31 \text{ K}$ . Based on the nozzle diameter therefore, the Reynolds number is  $Re = 635,308$ . The nozzle geometry resembles that of the experiments including the nozzle ring on which the actuators are mounted. The velocity profile at the entrance to the nozzle is that of a uniform flow (zero at the wall and  $U_{inlet}$  everywhere else). Perturbations were not introduced into the inflow due to the unknown perturbations in the experiment. Therefore, the

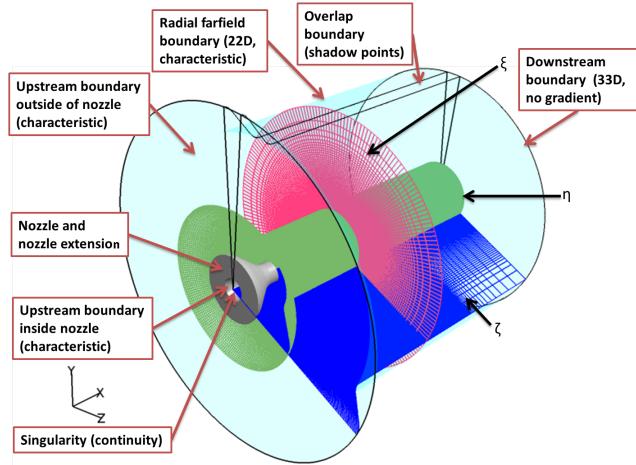


Figure 3: Computational domain

simulations have a laminar boundary layer at the nozzle exit while the experiments have a very thin turbulent boundary layer (the momentum thickness has been estimated to be  $0.09\text{mm}$ ). Previous studies have shown that despite this difference, the main features of the experimental observations are successfully reproduced by the computations.<sup>30,31</sup> Other studies have shown that a smaller 32 million point simulation is adequate to reproduce the features of the experiment.<sup>40</sup>

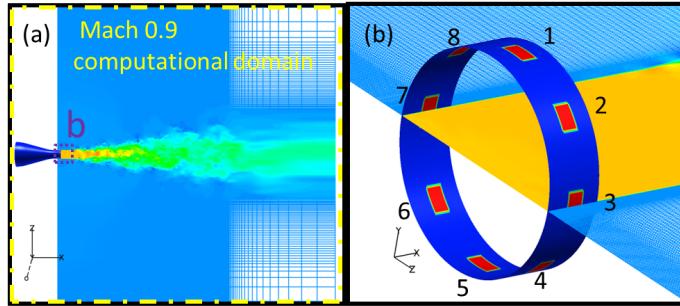


Figure 4: The computational domain including the nozzle (a), and the numerical actuator model (b)

The LAFFAs are modeled after the experiments using a surface heating technique to excite jet shear layer instabilities and azimuthal modes within the jet. Eight actuators are placed around the periphery of the jet on the nozzle collar at the locations and dimensions of the experiments as explained above in Section II.A. As shown in Fig. 4b, each actuator consists of a heated region of the nozzle wall which extends the azimuthal length corresponding to the separation distance between electrodes ( $3\text{mm}$ ) and has an axial extent equal to the length of the groove ( $1\text{mm}$ ). The temperature of the nozzle wall was assumed to be  $1.12T_\infty$ . When the actuator is on the temperature of the actuator region increases to  $5T_\infty$ . Little difference was seen in the previous work (Speth and Gaitonde<sup>41</sup>) for the temperature range measured in experiments (Utkin *et al.*<sup>42</sup>) for a Mach number of 1.3. The semi-empirical model is necessary to avoid first-principles simulation of the poorly understood plasma heating process, as well as to restrict the required computational resources to feasible levels (see Ref. 43).

Unlike acoustic drivers, the LAFFAs are on-off devices and thus can be represented by rectangular pulses with a duty cycle, which allows for a wide range of operation choices. Duty cycle is the percentage of actuator on time in an excitation cycle. Therefore, a duty cycle of 100% results in the actuator being on all the time. The experimental duty cycle varies with frequency, since the arc strike lasts a fixed time. Since the actuator model is empirical, the computational duty cycle was chosen to obtain similar control authority as in the experiment. This necessitates a higher duty cycle (10%) than the one used in the experiments (1.97% for

$St_{DF} = 0.25$ ). As noted earlier, despite the simplicity of the model, its success has been documented in Gaitonde and Samimy,<sup>30</sup> where, in addition to coherent structures, mean and fluctuating quantities have been compared. Furthermore, the mean flow structure with control was shown to match the theoretical predictions of Cohen and Wygnanski.<sup>44</sup>

Like the experiments, the axisymmetric ( $m = 0$ ) mode was employed to study a range of Strouhal numbers. The Strouhal numbers studied in the simulations include: 0.05, 0.15, and 0.25. Data was acquired every timestep at the point probes depicted in Fig. 2b as well as on several  $\xi$ ,  $\eta$ , and  $\zeta$  computational planes. Phase-averaged data were also computed for each of the simulations.

## IV. Results

### IV.A. Near-field Response to Excitation

Sinha *et al.*<sup>26</sup> previously found that each pulse from the actuators produced a perturbation, which would amplify, roll into a large-scale structure, then the structure would grow, saturate, and decay as it advected through the jet shear layer. In the irrotational near-field, the signature of these large-scale structures took the form of a compact waveform, with a strong compression wave followed closely by a strong expansion wave. At low enough excitation frequencies (e.g.  $St_{DF} = 0.05$ ), the characteristic period of this compact waveform is much less than the excitation period. Hence, the structures seeded by the excitation do not interact with one another as they evolve downstream. Because these structures evolve independently, their behavior can be thought of as representing the response of the jet to a single perturbation: the impulse response of the jet. As the excitation Strouhal number is increased, the excitation period will decrease to the point where the characteristic period of the impulse response is on the same order as, or greater than, the excitation period. As the period of actuation approaches the characteristic period of the impulse response, the waveforms extracted by the phase-averaging technique are largely unmodified from that of the impulse response; they are simply spaced closer together. As the Strouhal number is increased beyond this initial interaction Strouhal number, the phase averaged waveform resembles a sine-wave ( $St_{DF} = 0.25$ ).

This behavior of independent evolution and periodic interactions between structures can be observed in Figure 5, which depicts the phase averaged waveforms for the experiments and computations at an axial distance of  $3D$  on the first array position ( $r/D = 1.5$ ) for different excitation frequencies. The impulse response of the jet is observed at the lowest excitation frequency,  $St_{DF} = 0.05$ , as the characteristic period of the compression and expansion waves is much less than the period of excitation. For the experimental  $St_{DF} = 0.15$  and 0.25, the magnitudes of the peak and trough are nearly unchanged and the shape is largely unaffected, yet the characteristic period of the response is reduced due to the structure interactions. Further increases in the excitation Strouhal number, to  $St_{DF} = 0.35$  and 0.50, yields structures for which the amplitude has been significantly reduced, as has the characteristic period.

Preliminary simulation results are shown in Fig.5(b). Overall, the main qualitative features are reproduced, though clear quantitative differences are evident. Given that the receptiveness of a shear layer to perturbations is highly sensitive to changes in shear layer characteristics and actuator locations, exact quantitative matches between the experimental and numerical database is not expected. Nonetheless, the simulations show the same compact evolution of peaks and troughs as in the experiment. Previous simulation results with different excitation frequencies have been presented in Ref.<sup>40</sup> for a Mach 1.3 jet. There too, the lower frequencies yield structures that do not interact with each other, resulting in a long “quiet” time between peaks and troughs. As the frequency is increased, interactions between successive events results in a more continual variation of the phase-averaged response. The empirical model is currently being adjusted to aid in a more accurate quantitative comparison,

A simple metric for evaluating the growth, saturation, and decay of the large-scale structures generated by the excitation is the mean-square of the pressure fluctuations; these are shown in Fig. 6 along the first pressure probe array for the different excitation Strouhal cases and the baseline (0.00) no-control case. Again, the  $P_{ms}$  profiles show the same trends in computation and experiment, though quantitative differences exist for each Strouhal number, and as the frequency is varied. The baseline no-control case exhibits the lowest mean square values followed by the lowest excitation Strouhal number. The mean square pressure increases with increasing excitation Strouhal number until the column mode Strouhal number is reached ( $St_{DF} \approx 0.3$ ) at which point the mean square pressure starts to decrease with increasing excitation Strouhal number. Although the computations do not consider Strouhal numbers higher than the most amplified column mode value, we note that in an earlier numerical study a similar reduction in control authority was observed at

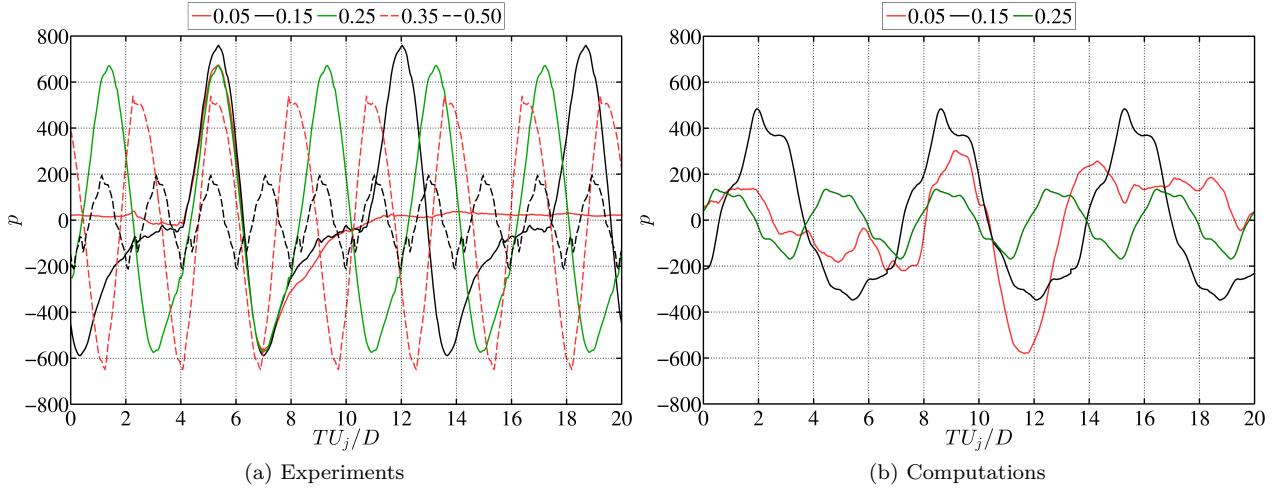


Figure 5: Phase-averaged waveforms at  $x/D = 3$ ,  $r/D = 1.5$

higher excitation frequencies<sup>31</sup> and the near field pressure fluctuations were also diminished.<sup>45</sup> For both experiments and simulations, increasing the excitation Strouhal number also yields an upstream shift in the saturation location. These results are consistent with those of other researchers, who have shown that perturbations of higher frequencies saturate earlier upstream than lower frequencies.<sup>46,47</sup> Overall therefore, despite quantitative differences between experimental and numerical databases, expected given the nature of the uncertainties associated with the nozzle boundary layer and the precise effect of the plasma actuator, there is sufficient indication that the main features are in fact similar in both approaches. This greatly facilitates combined use of both approaches, leveraging of the strengths of each, to obtain a much higher level of insight than with one technique alone.

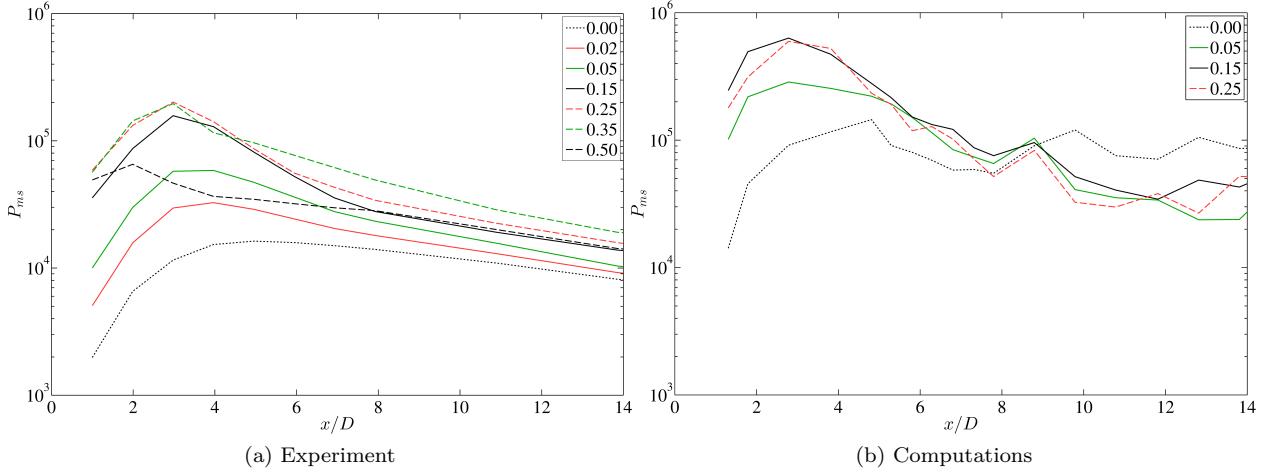


Figure 6: Mean-square pressure along the first array (first probe located at  $r/D = 1.2$ )

#### IV.B. Vortex Dynamics

Since the simulations reproduce the main features observed in the experiment, the three-dimensional unsteady datasets are employed to study the large scale structures associated with the near-field dynamics. The 3-D structure dynamics that produce the intricate waveforms in Fig. 5 can not be easily obtained in experiments, due to the lack of time resolved volumetric PIV. Therefore, the LES results are used to analyze the structures producing the waveforms.

For each excited case, Fig. 7 shows phase-averaged isolevels of Q-criterion ( $Q = 0.35$ ) colored by axial

velocity with a background of dilatation in gray scale. Each figure depicts two phases of the excitation period ( $\phi = 0.1(2\pi)$  and  $0.6(2\pi)$ ). At each phase, the locations  $x/D = 2$  and  $4$  of the first array are labeled. For the  $St_{DF} = 0.05$  cases (Fig. 7a), the A' and A structures are depicted at phase  $0.1(2\pi)$ . At phase  $0.6(2\pi)$ , the structure observed during the first phase has already broken up resulting in no observable actuator induced structures at this phase for this Strouhal number.

The high frequency cases develop rollers due to the excitation that grow and interact with other actuator induced structures as they propagate downstream. Evidence for this has also been presented in the experiments by Sinha *et al.*<sup>26</sup> Initially, the structures that are produced at phase  $\phi = 0.6(2\pi)$  of Figs. 7b and c are similar to those associated with the impulse response (Fig. 7a). However, as each structure grows and propagates downstream, it interact with the structure generated during the prior actuator on event. Thus structures B and B' are equivalent to A and A' respectively, but belong to the previous/subsequent actuator pulse.

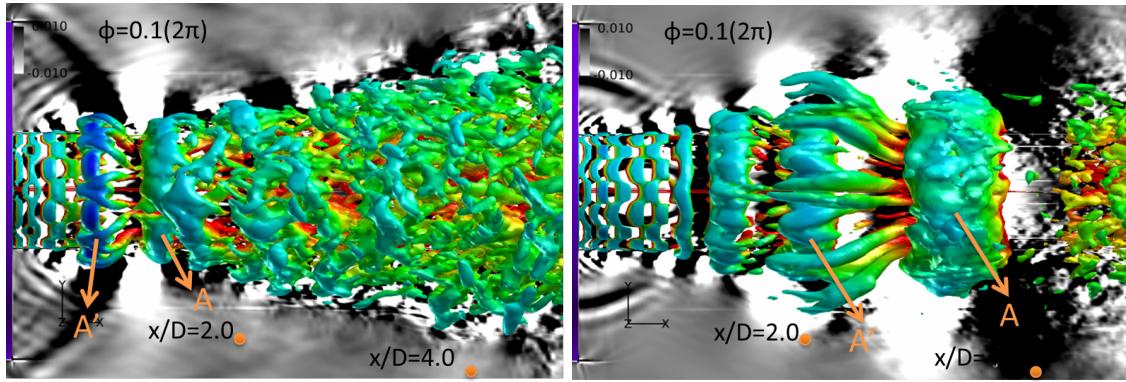
Figure 7b depicts results with the  $St_{DF} = 0.15$  case. Here the structures start to interact at an axial distance of about  $4D$ . At  $\phi = 0.1(2\pi)$ , phase the characteristic structures seen in Figs. 7a are observed again. Half a phase later A' is broken up close to  $x/D = 4$  while the ill formed B structure continues to interact with the remnants of A'. B' and B are similar structures to the ones denoted in the impulse case (Fig. 7a) however structure B is not as well formed as structure A. Conversely, B' is more robust than the previous A' structure. This indicates a degree of feedback response of the structures between each excitation pair.

Figure 7c depicts the isolevels of the high frequency ( $St_{DF} = 0.25$ ) case. Successive structures begin to interact even earlier, starting at an axial distance of  $2D$ . Since, the reaction to the actuation is cyclic, the structures seen at the end of the potential core at one phase ( $\phi = 0.1(2\pi)$ ) begin to develop at similar phases in the next cycle. Structure B/A' in phase  $\phi = 0.1(2\pi)$  occurs when structures B and A' influence each other through self and induced effects. This compression occurs due to the relatively high convective velocity of B (which is closer to the nozzle exit where the speed is higher) compared to A'. This interaction is quasi-linear, creating a sine-like response in the near-field pressure through linear superposition of the two actuator structures (B and A'). This quasi-linear superposition effect has previously been documented by Sinha *et al.*<sup>26</sup> and Crawley *et al.*<sup>48</sup>

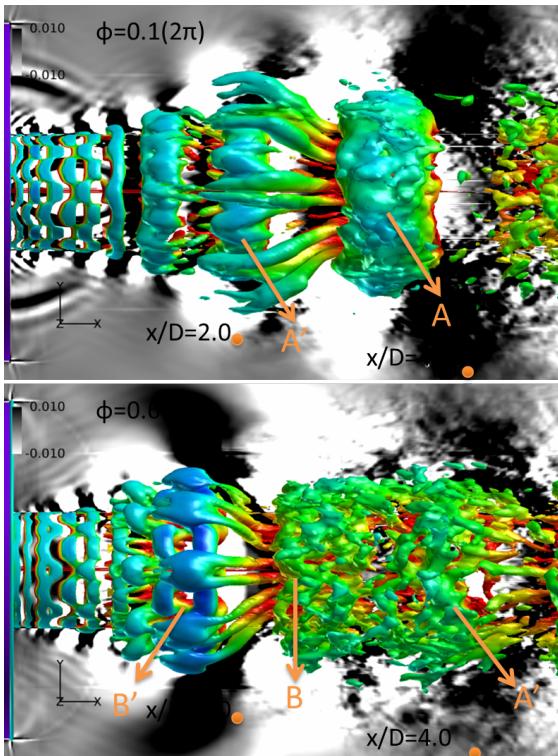
The structures affect the near-field as seen by the strong dilatation waves surrounding each large scale structure. The dilatation values of Fig. 7 may be connected to the pressure profiles of Fig. 8a. The white dilatation waves (lower dilatation value) correspond to an increase in pressure while the black dilatation waves (higher dilatation value) correspond to a decrease in pressure. These pressure fluctuations can be readily seen in the phase-averaged pressure probes along the first array in Fig. 8a for Strouhal numbers of 0.05 and 0.25. At a phase of  $\phi = 0.1(2\pi) = 36^\circ$  in Fig. 7c, the  $x/D = 4$  point probe is entering a white dilatation region (increase of pressure) while the  $x/D=2$  probe is entering a black dilatation wave corresponding to a decrease in pressure. In Fig. 8a, an increase of pressure is seen at  $x/D = 4$  and a decrease is seen at  $x/D = 2$  for the phase of  $36^\circ$ .

Similarly, the correlations between the points on the first array can also be related to the phase-averaged structures. Figure 8b portrays the two-point correlations of pressure comparing  $x/D = 4$  to  $x/D = 2$  on the first array for all simulated cases. Each Strouhal number exhibits correlation peaks which are one actuation period apart. The  $St_{DF} = 0.05$  case exhibits a damped sine wave which has a large peak at about  $\Delta\phi = -0.15(2\pi)$  and exhibits decay on either side until the correlation begins to grow again due to the next actuation pulse. While the  $St_{DF} = 0.05$  case has low correlation values between actuation periods, the high frequency cases exhibit a more sine-like pattern due to the interacting structures. These correlations can be readily seen in the phase-averaged results of Fig. 7. In Fig. 7b, at  $\phi = 0.1(2\pi)$ ,  $x/D=4$  is experiencing a dark dilatation wave (decrease in pressure). A half an excitation period later ( $\phi = 0.6(2\pi)$ ),  $x/D = 2$  is also experiencing a dark dilatation wave. This translates to a positive correlation in Fig. 8b at a time difference of half a cycle. Also for a  $St_{DF} = 0.25$ , the phase of  $0.1(2\pi)$  exhibits a rising pressure region at  $x/D = 4$  and thereby half a phase later ( $\phi = 0.6(2\pi)$ ) the  $x/D=2$  location is also experiencing a rise in pressure which is indicated as a positive correlation in Fig. 8b for a  $\Delta\phi = 0.5(2\pi)$ .

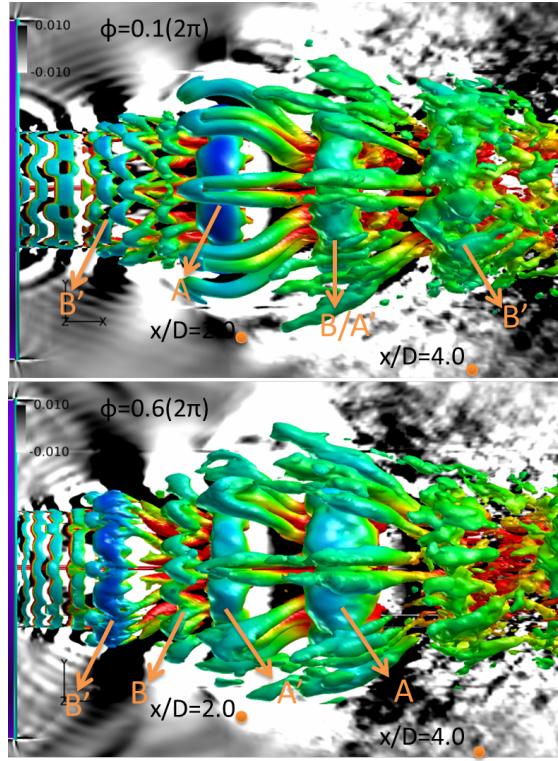
To further understand the dynamics of the structures and how they relate to the near-acoustic-field, spectral POD will be employed together with wavelet decomposition (discussed further in Section IV.D). First, the shear layer and near-field will be decomposed using spectral POD to validate and confirm the results from other spectral POD decompositions present in literature.<sup>2,18,21</sup> Then, the wavelet decomposed data (separation of the acoustic and hydrodynamic components) of the near-field will be further decomposed into the spectral modes to evaluate the relevant processes in the acoustic signal.



(a)  $St_{DF} = 0.05$

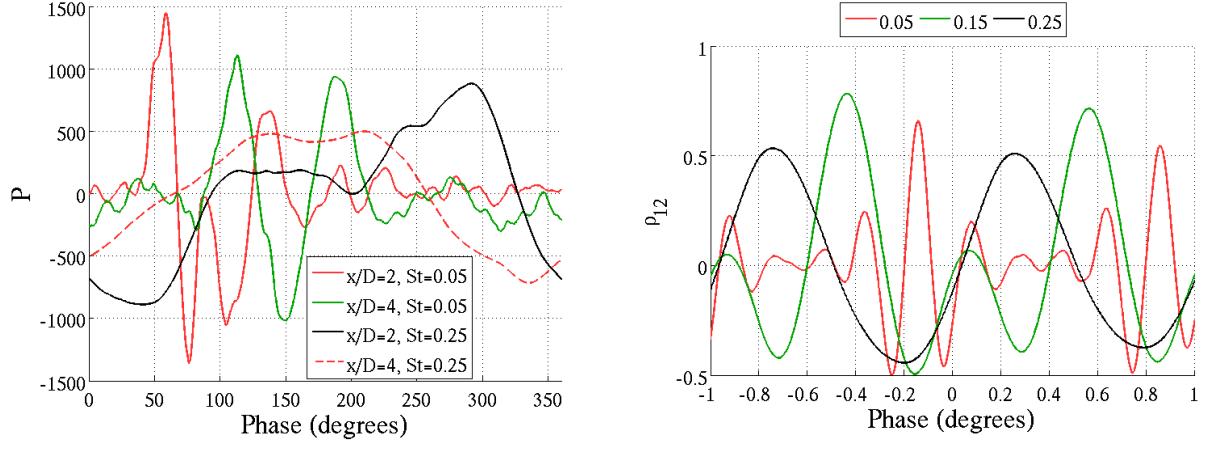


(b)  $St_{DF} = 0.15$



(c)  $St_{DF} = 0.25$

Figure 7: Simulations of the phase averaged iso-levels of Q-criterion colored by axial velocity with gray scale of dilatation



(a) Phase averaged point probes on the first array for  $St_{DF} = 0.05$  and  $0.25$   
(b) Two-point correlations of  $x/D=2$  to  $x/D=4$  for each excitation Strouhal number

Figure 8: Resultant near-field dynamics due to large scale structures

#### IV.C. Far-field Response to Excitation

The behavior of independent and periodic interaction of the jet response to excitation is not limited exclusively to the hydrodynamically-dominated regions of the jet, but in fact holds for the acoustic far-field as well, at least at angles close to the jet axis. This can be observed in Figure 9, where the phase-averaged response of the jet in the acoustic far-field at a polar angle of  $30^\circ$  (with respect to the downstream jet axis) has been plotted for the experimental jet. For legibility, only a select number of excitation Strouhal numbers have been included. As with the irrotational near-field, the acoustic far-field exhibits a compact waveform for the lowest excitation Strouhal numbers. For  $St_{DF} = 0.15$  and  $0.25$  the primary expansion and compression waves remain nearly unchanged from the fundamental response, aside from a slight augmentation of the peak of the compression wave. However due to the periodic excitation, the weaker expansion and compression waves are no longer identifiable, as they are subsumed by the primary waves. At higher  $St_{DF}$ , a continuous oscillation between sharp expansion and compression waves is again observed, though both the amplitude and period are reduced from the impulse response. As before, it was found that a linear superposition of the impulse response can well predict the waveform shape and amplitude at the higher excitation frequencies.

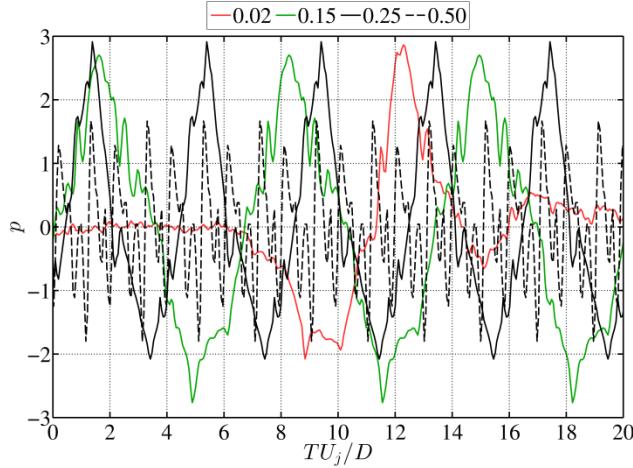


Figure 9: Phase-averaged waveforms in the acoustic far-field at  $30^\circ$  for select excitation frequencies, in the experimental jet.

## IV.D. Near-field Hydrodynamic and Acoustic Decomposition

### IV.D.1. Decomposition Method

In this work, a wavelet-based decomposition method, new to the field of aeroacoustics, is utilized in order to decompose the raw near-field pressure into its hydrodynamic and acoustic components. Use of a multidimensional, continuous wavelet transform to extract intermittent events with a given spatio-temporal character is not immediately straightforward, due to the global nature of the scale factor. A speed-tuning parameter was introduced to the wavelet transform (now specifically referred to as a spatio-temporal wavelet transform) by Antoine *et al.*<sup>49</sup> for use in motion tracking and identification. Following the work of Kikuchi and Wang,<sup>50</sup> the definition for the daughter wavelets is modified to be, in two dimensions  $(x, t)$ ,

$$\psi_d(x, t; x', t', s, c) = s^{-1} \psi \left( s^{-1} c^{-1/2} (x - x'), s^{-1} c^{1/2} (t - t') \right) \quad (2)$$

From this, the spatio-temporal wavelet coefficients can be computed as

$$\tilde{p}(x', t', s, c) = \int_{-\infty}^{\infty} \int_{-\infty}^{\infty} p(x, t) \psi_d^*(x, t; x', t', s, c) dx dt \quad (3)$$

In practice however, it is usually far more computationally efficient to compute Eq. 3 in Fourier space (by the convolution theorem) and then transform the coefficients back into physical space. This leads to an alternative interpretation of the wavelet transform, that of a series of bandpass filters, the passband envelope, centroid, and width being dictated by the scale, speed, and mother wavelet.<sup>49,51,52</sup> The decomposed signals can now be reconstructed as

$$p_h(x, t) = \frac{1}{C_\delta} \int_{0^+}^{a_\infty^-} \frac{dc}{c} \int_{0^+}^{\infty} \frac{ds}{s^2} \tilde{p}(x, t, s, c) \quad (4)$$

and

$$p_a(x, t) = \frac{1}{C_\delta} \int_{a_\infty}^{\infty} \frac{dc}{c} \int_{0^+}^{\infty} \frac{ds}{s^2} \tilde{p}(x, t, s, c) \quad (5)$$

The constant factor  $C_\delta$  serves as an energy scaling, and appears because we are reconstructing the signal using a different analyzing wavelet (in this case, a delta function) than the mother wavelet used in the forward transform.<sup>51,52</sup>

As with the Fourier-based decompositions, the wavelet-based decompositions are performed along each radial microphone array position, separately. Similar preprocessing of the raw signal (that is, the application of the Tukey window along the temporal dimension, zero-padding along the spatial dimension, and cubic interpolation onto a regularly spaced axial grid) was performed in order to reduce the spectral leakage inherent in the DFT (as the wavelet transform was computed in the Fourier domain). The reconstruction was then performed only over those sections of the raw signal which were not amplitude-modulated by the application of the window. In the current work, the (1+1) dimensional Morlet wavelet was chosen as the mother wavelet:

$$\psi(x, t) = e^{i(k_0 x + \omega_0 t)} e^{-(x^2 + t^2)/2} \quad (6)$$

which the reader will recognize as simply a plane wave modulated by a Gaussian. Though simplicity was a factor in this decision, previous results analyzing phase-averaged waveforms in the far-field found acoustic emissions with a characteristic waveform that share some resemblance to the Morlet wavelet.<sup>48</sup> The base oscillation frequencies,  $(k_0, \omega_0)$  were set to  $(\pm 6, 6)$  (the dual sign for  $k_0$  being necessary to recover both forward and backward traveling waves), and  $\hat{\psi}(k, 0) = 0$  and  $\hat{\psi}(0, \omega) = 0$  so as to ensure that the mother wavelet met the admissibility criterion.

### IV.D.2. 2D Experimental Decomposition

Before analyzing the separate fields in the context of the noise generation problem, the efficacy of the linear wavenumber-frequency filtering methodology used in this study will briefly be evaluated. Results for the decomposition of the unforced jet can be found in Figure 10, where the power spectral densities for the total, hydrodynamic, and acoustic waveforms in the unforced jet have been plotted for  $x/D = 8$ ,  $y/D = 2.2$ . Three vertical lines have been overlaid on the plot, corresponding to the critical frequency, and the far-field spectral peak at  $30^\circ$  and  $90^\circ$ , respectively. The critical frequency denotes the frequency at which the near-field spectra transition from hydrodynamically-dominated to acoustically-dominated, and can be visually

identified by a change in the slope of the spectral decay. This frequency has been found to scale as  $fy/U_c$ , resulting in a consistent value of unity.<sup>26</sup> The convective velocity was estimated using two-point correlations between successive microphones in the upstream region of the jet, and was found to be  $0.69U_j$ .

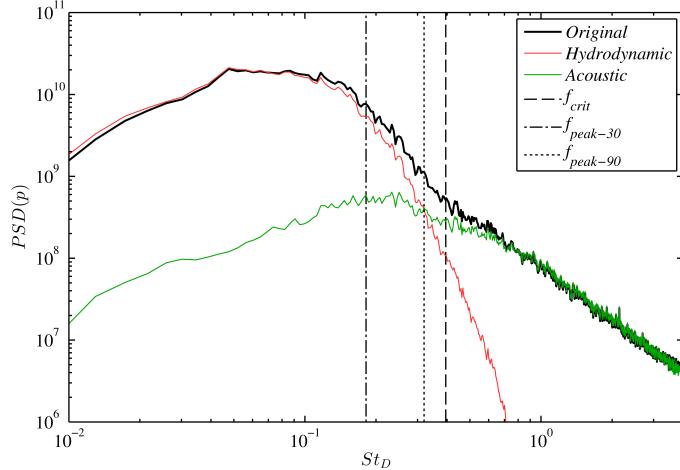


Figure 10: Spectra of the raw and decomposed fields in the unforced jet at  $x/D = 8$ ,  $y/D = 2.2$ .

As expected, the hydrodynamic component matches well with the total signal in the low frequency portions of the spectra, while the acoustic component matches well in the high frequency portions. At both positions, the critical frequency accurately predicts the frequency at which the cross-over in amplitude between the hydrodynamic and acoustic components occurs. Though not shown here for brevity, a change in shape in the acoustic spectra is evident with probe position relative to the end of the potential core. At locations downstream of the end of the potential core and at low angles with respect to the jet axis, the spectrum is more peaky (reminiscent of far-field spectra at aft angles) and the acoustic spectral peak frequency matches well with the far-field peak at  $30^\circ$ . At locations corresponding to high angles the acoustic spectrum is more broadband and rounded (reminiscent of the well-known shape of the far-field spectra at sideline angles) and the acoustic peak frequency matches well with the far-field peak at  $90^\circ$ . Overall, the results found here lend strong credence to the argument that the wavelet-based filtering is producing a realistic reconstruction of the hydrodynamic and acoustic fields in both the forced and unforced jets, and accurately capturing the dynamics of the large-scale structures and their radiated noise. Currently, work is underway in order to extend the algorithm to higher-dimensions for use with the numerical database.

## V. Future Work

The work to date has begun to shed light on the importance of the large-scale structure interactions in determining the acoustic far-field: the structure-structure interactions (or lack thereof) appear to govern the streamwise evolution of the structures which in turn governs the acoustic emission. What remains to be done, is directly linking the relevant vortex dynamics of the large-scale structures to the acoustic emission events, and in the process identifying a simplified aeroacoustic source mechanism. To this end, the information provided separately in the experimental and numerical databases complement one another in providing a more complete representation of the acoustically-relevant dynamics of the large-scale structures. Additional comparisons between the experimental and simulation databases can be made in terms of vortex interactions, shortening of the potential core and spreading of the shear layer, as well as other statistics such as turbulent kinetic energy.

The final paper will include updates on extensions of the present lines of inquiry. As already mentioned, phase-locked PIV data will be obtained in the experimental jet, allowing both qualitative and quantitative measures of the large-scale structure evolutions (growth, saturation, and decay, structure-structure interactions, shear layer spreading rate, potential core length) to be evaluated from the instantaneous and phase-averaged velocity profiles with the aid of standard structure identification techniques. The addition of the decomposed LES database provides the ability to compute the full source term identified in Lighthill's

acoustic analogy, as well as simplified models such as the compressible equivalents of the Lamb vector divergence and TKE Laplacian, identified by Cabana *et al.*<sup>53</sup> as important in the sound production process in a simulated mixing layer. The structure of the radiating field can be analyzed in both the physical and wavenumber domains. Finally, spectral POD and azimuthal Fourier mode decomposition will be performed on the decomposed (acoustic and hydrodynamic) signals of the simulations in order to better understand the mechanisms that produce the acoustic waves and how these waves grow/decay, and to relate the results of the excited jet to those of the natural jet.

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