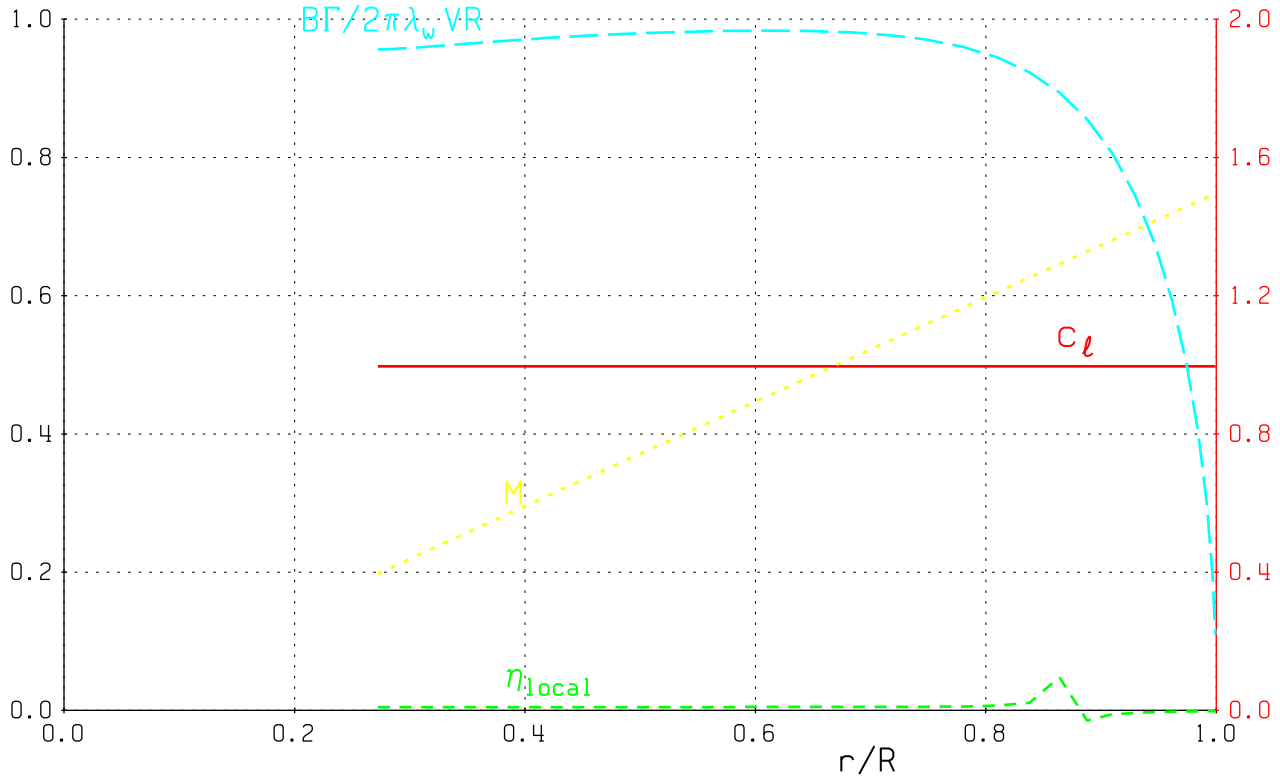
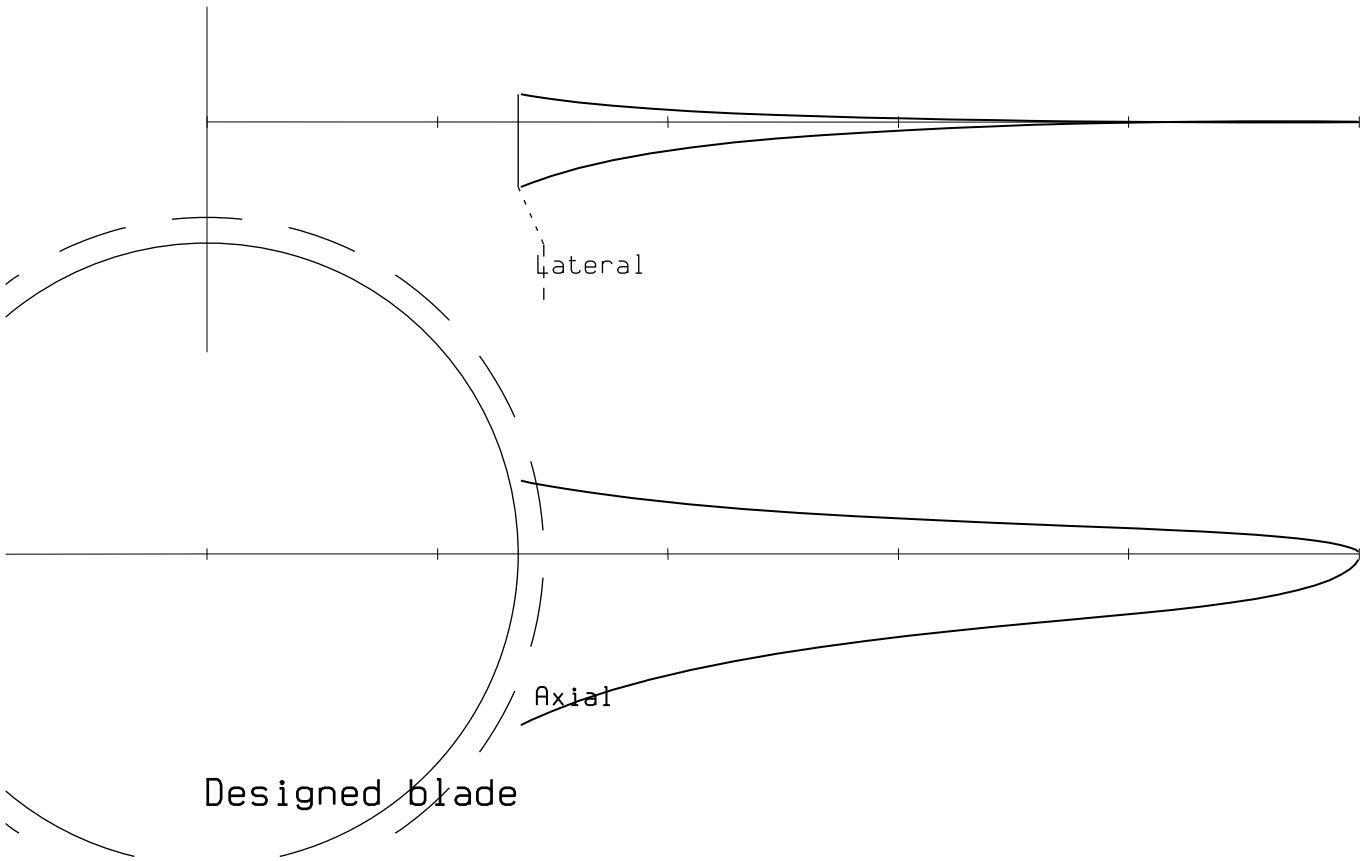
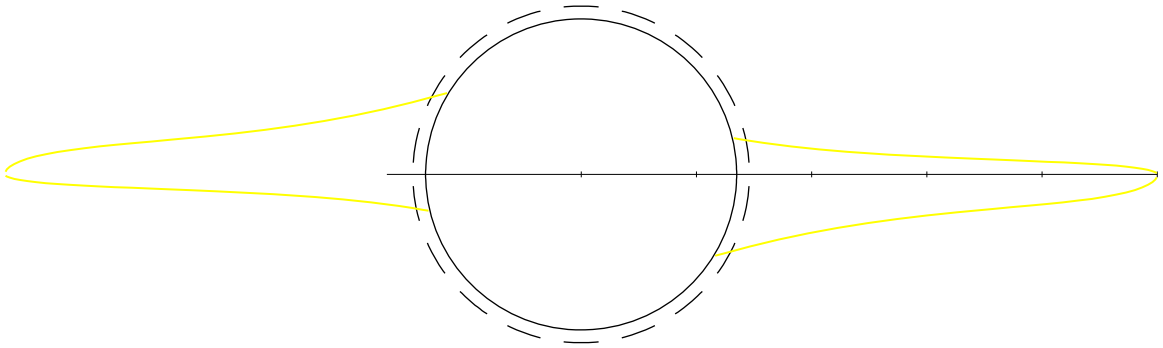
Designed blade

#bld= 2	R m = 0.450	$\sigma_{3/4}$ = 0.0518	β_{twist}= 22.489	
Vm/s= 0.100	V/ΩR= 0.0004	P_C =	C_P = 0.0060	η_{ideal}= 0.0062
h km= 0.000	J = 0.0012	T_C =	C_T = 0.0625	η = 0.0128
T kN= 0.4087	P kW= 3.1970	RPM = 5411.3	β_{tip} = -1.718	
Helicopter	C_{TH} = 0.008060	C_{PH} = 0.000247	C_{TH}/σ= 0.1556	FOM = 2.0697





Designed blade

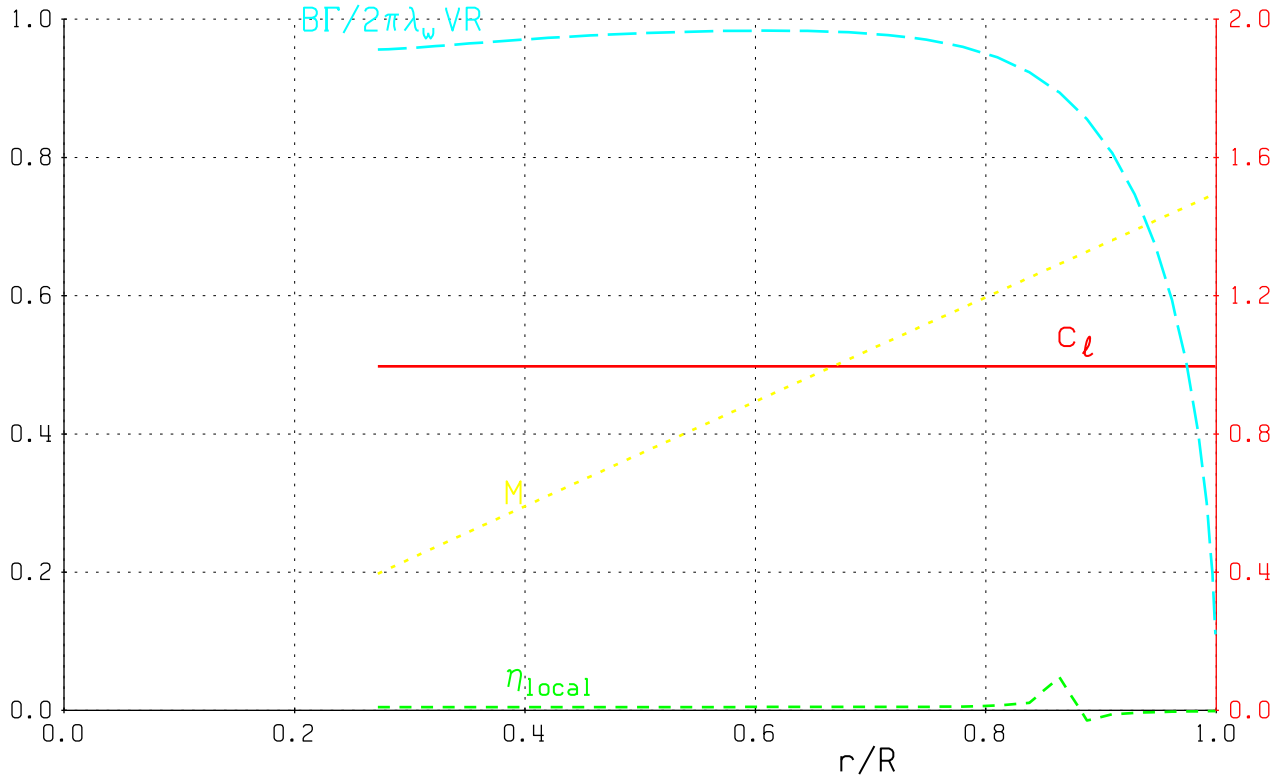


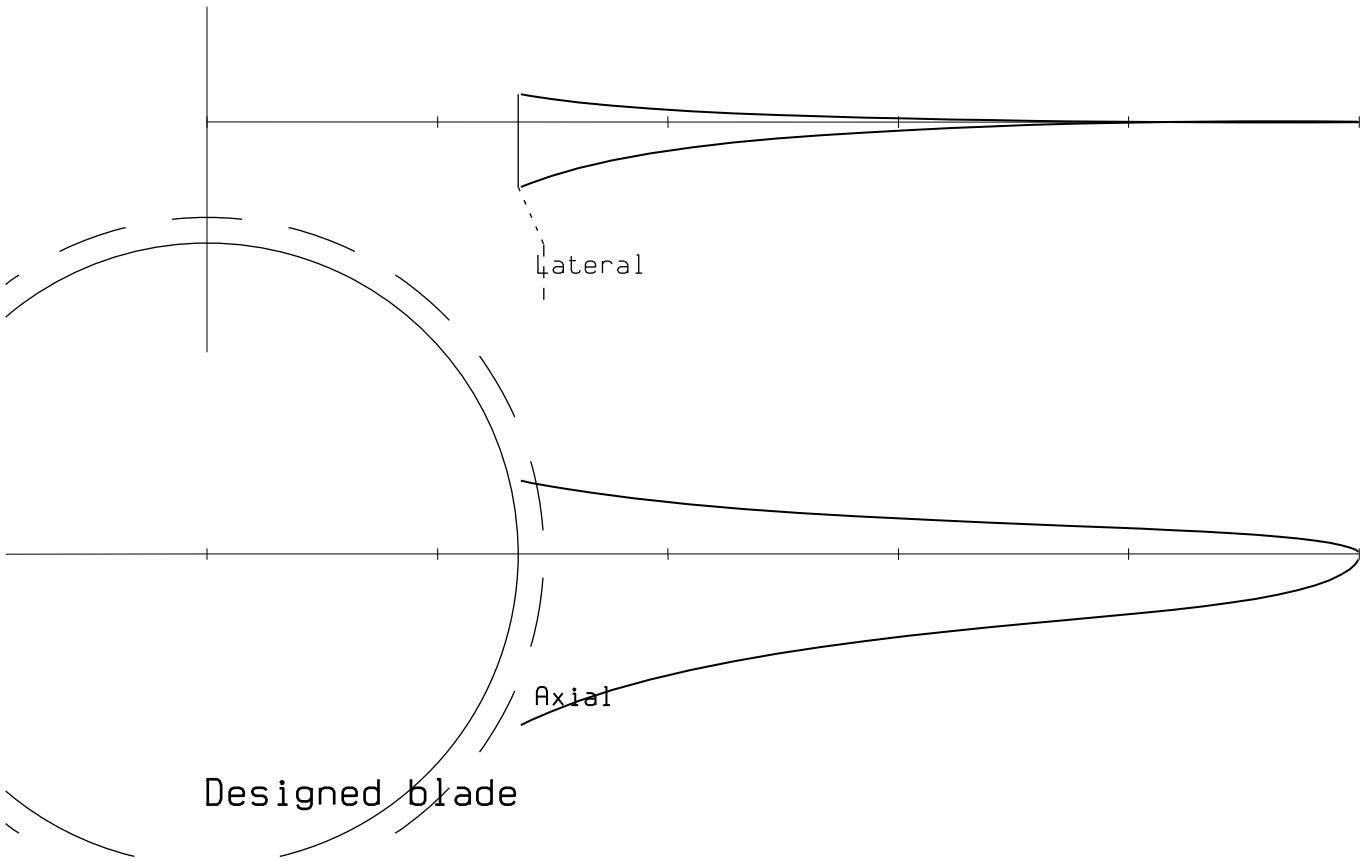
Designed blade

#bld	= 2	R m	= 0.4500	A m²	= 0.589796
$\sigma_{3/4}$	= 0.0518	R_{hub}	= 0.1215		
β_{twist}	= 22.489	R_{wak}	= 0.1315		

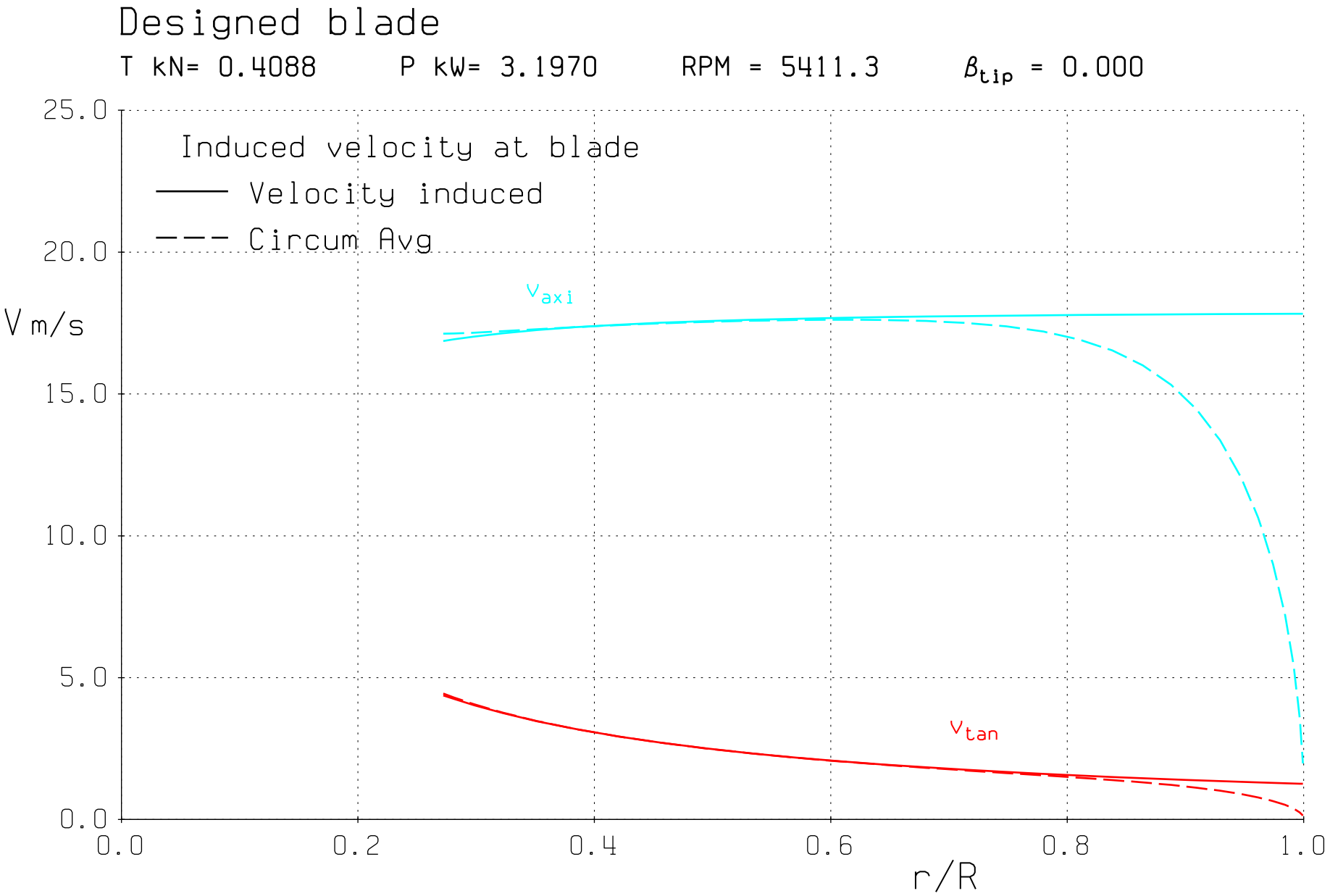
Designed blade

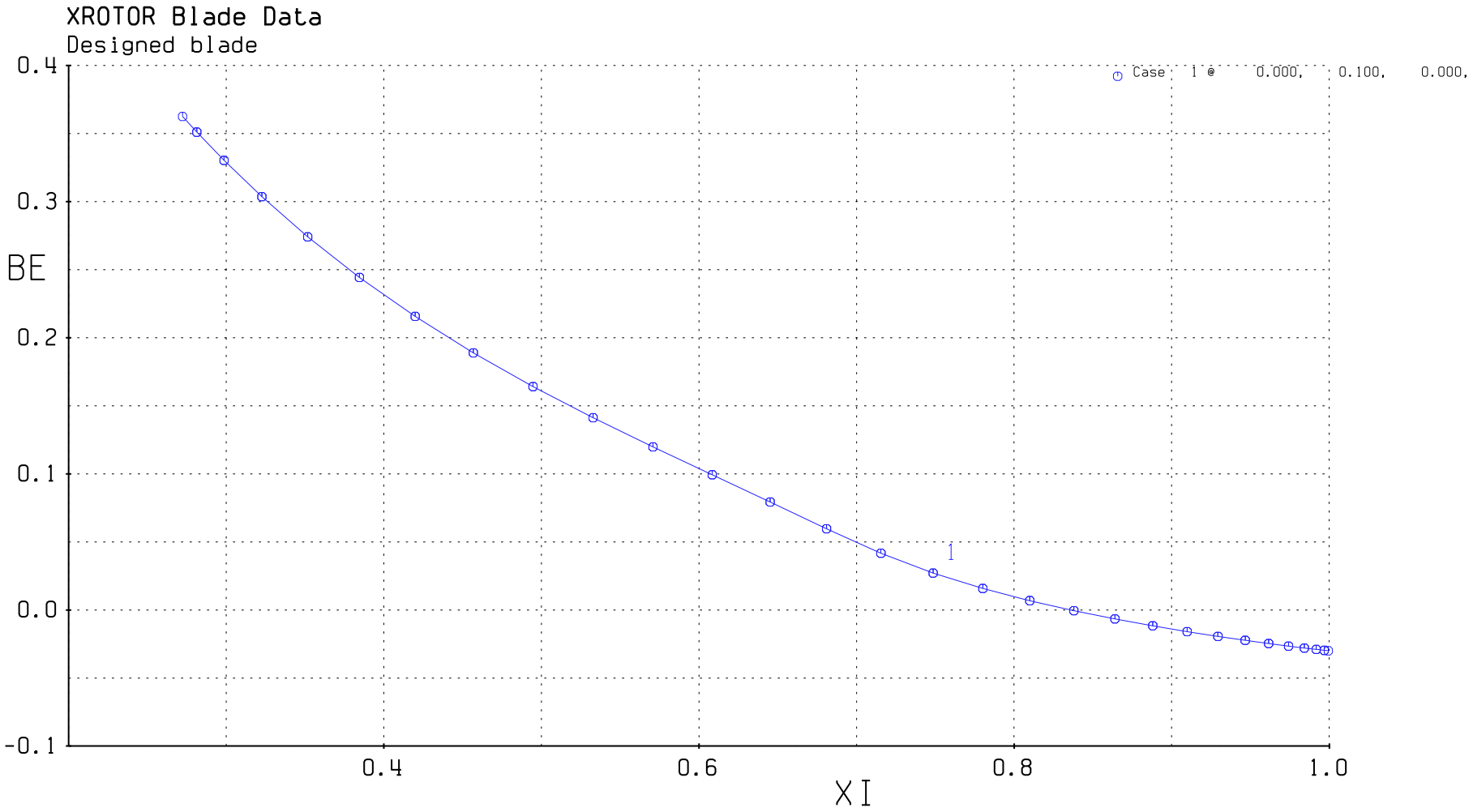
#bld= 2	R m = 0.450	$\sigma_{3/4}$ = 0.0518	β_{twist}= 22.489	
Vm/s= 0.100	V/ΩR= 0.0004	P_C =	C_P = 0.0060	η_{ideal}= 0.0062
h km= 0.000	J = 0.0012	T_C =	C_T = 0.0625	η = 0.0128
T kN= 0.4088	P kW= 3.1970	RPM = 5411.3	β_{tip} = -1.718	
Helicopter	C_{TH} = 0.008060	C_{PH} = 0.000247	C_{TH}/σ= 0.1556	FOM = 2.0697

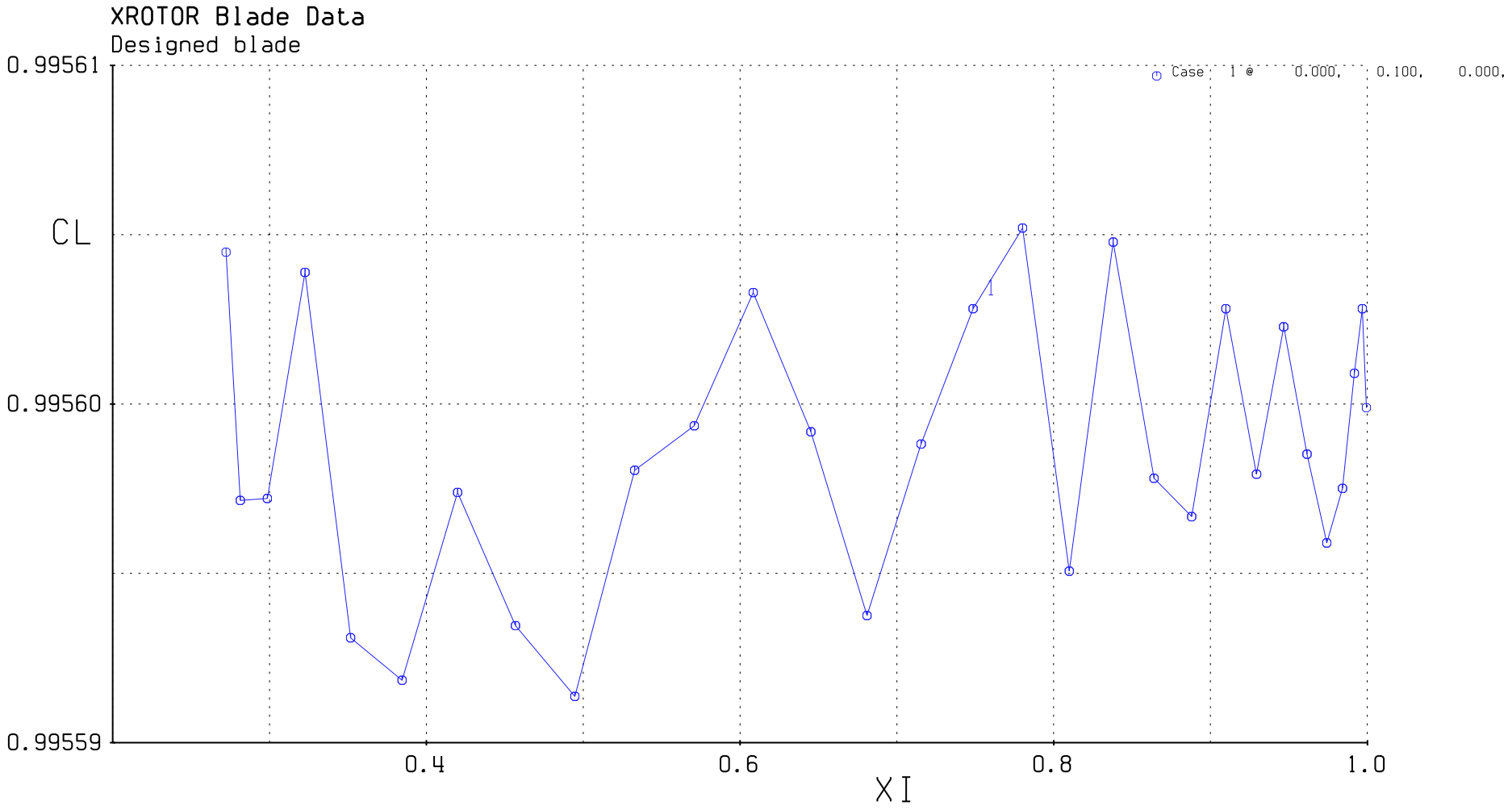


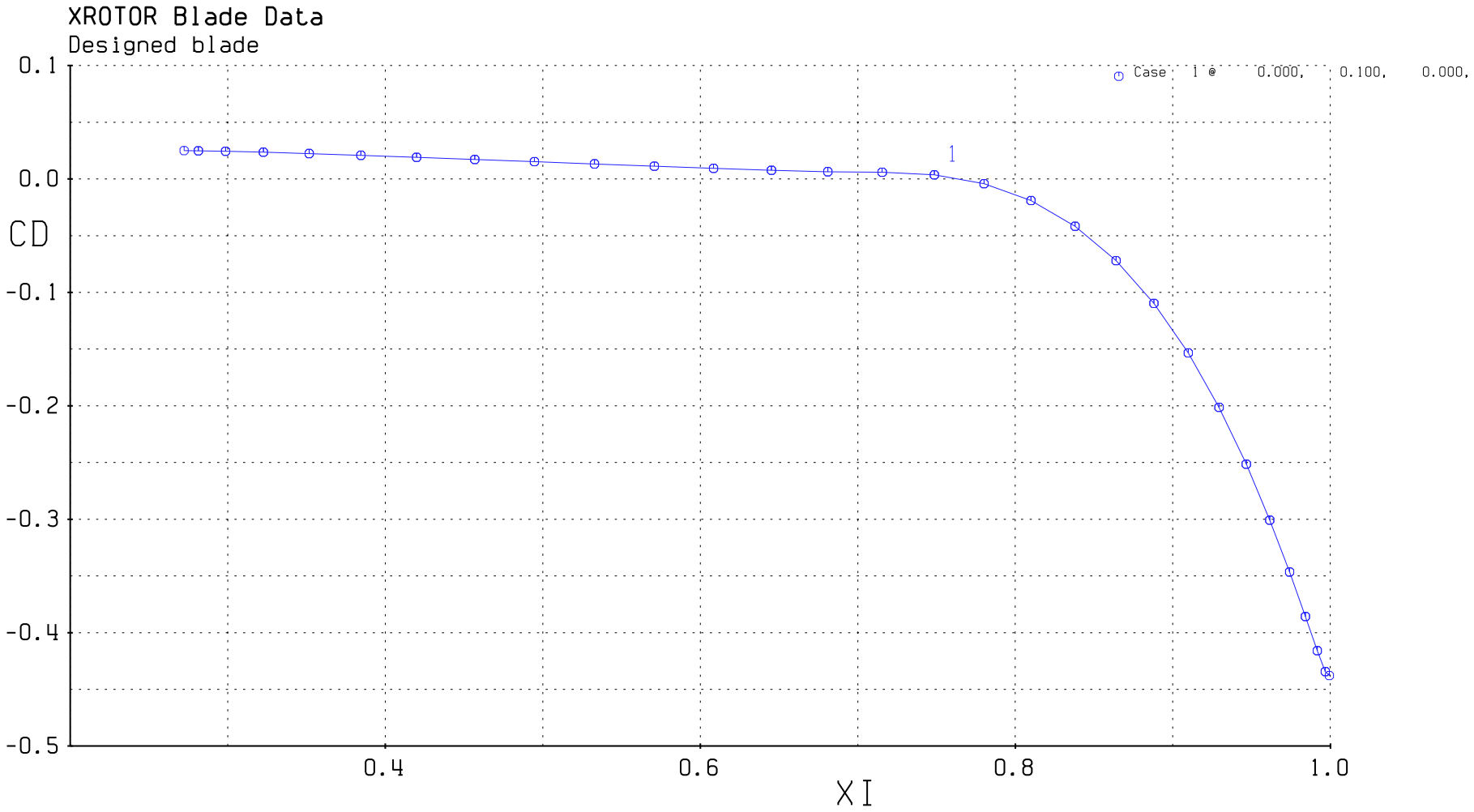


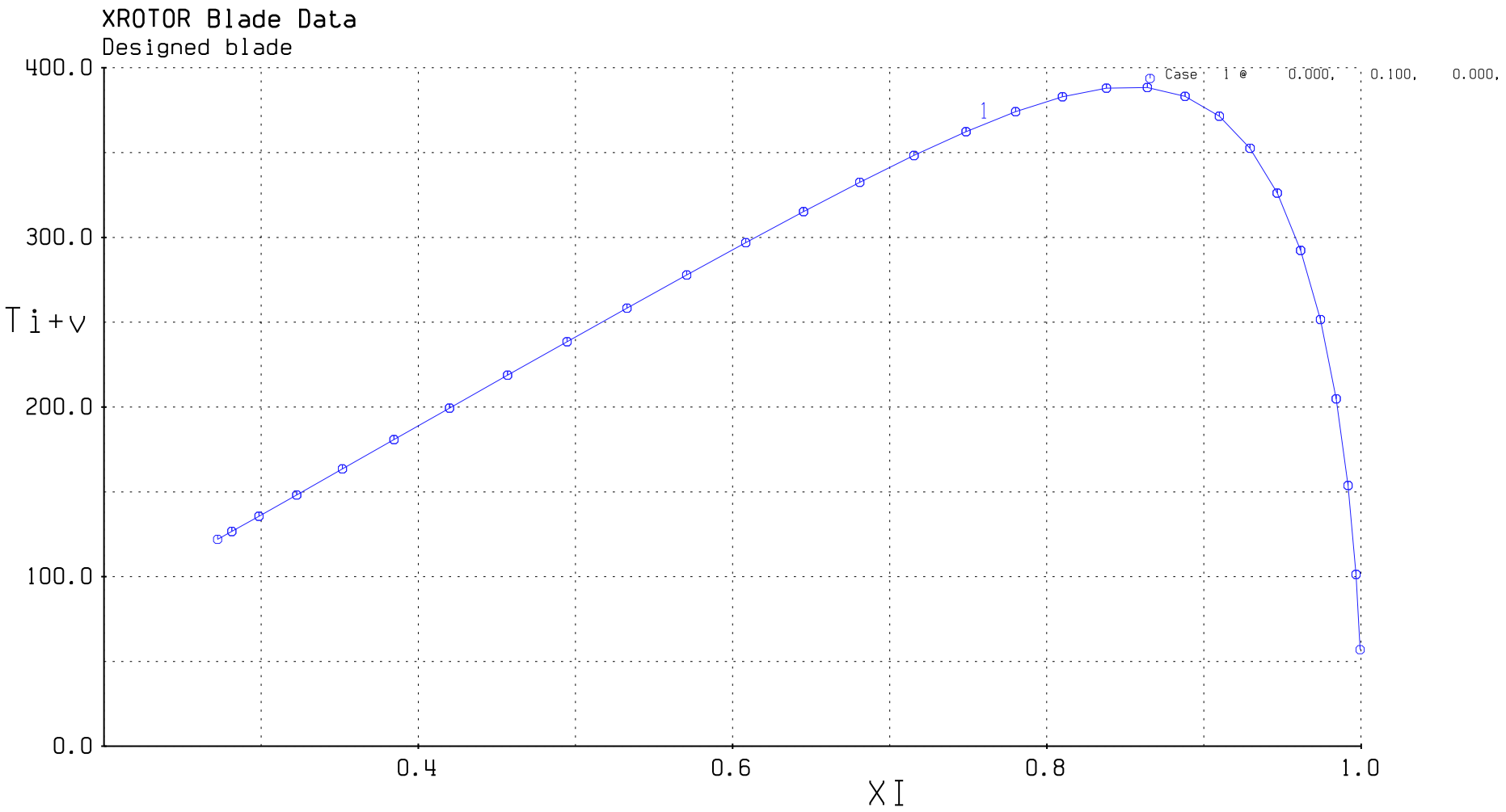
Designed blade

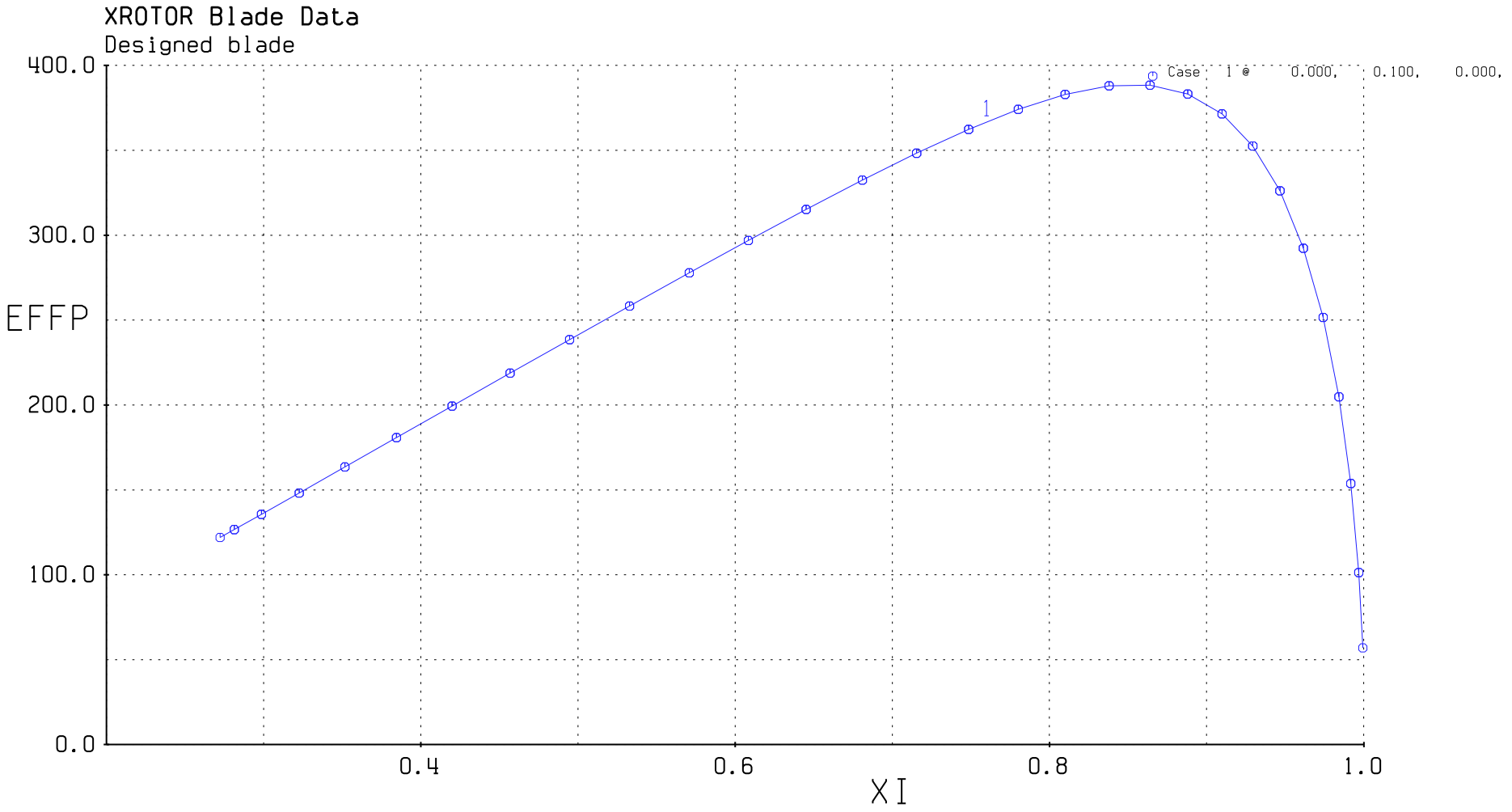

Designed blade
T kN= 0.4088
P kW= 3.1970
RPM = 5411.3
β tip = 0.000
Induced velocity at blade
Velocity induced
Circum Avg
V axi
V tan
V m/s
r/R
25.0
20.0
15.0
10.0
5.0
0.0
0.0
0.2
0.4
0.6
0.8
1.0


XROTOR Blade Data
Designed blade
Case 1 @ 0.000, 0.100, 0.000,
0.4
0.3
0.2
0.1
0.0
-0.1
BE
0.4
0.6
0.8
1.0
XI
1


XROTOR Blade Data
Designed blade
Case 1 @ 0.000, 0.100, 0.000,
0.99561
CL
0.99560
0.99559
0.4
0.6
0.8
1.0
XI


XROTOR Blade Data
Designed blade
Case 1 @ 0.000, 0.100, 0.000,
1
0.1
0.0
-0.1
-0.2
-0.3
-0.4
-0.5
CD
0.4
0.6
0.8
1.0
XI


XROTOR Blade Data
Designed blade
Case 1 @ 0.000, 0.100, 0.000,
400.0
300.0
200.0
100.0
0.0
Ti+v
0.4
0.6
0.8
1.0
XI


XROTOR Blade Data
Designed blade
Case 1 @ 0.000, 0.100, 0.000,
400.0
300.0
200.0
100.0
0.0
EFFP
0.4
0.6
0.8
1.0
XI