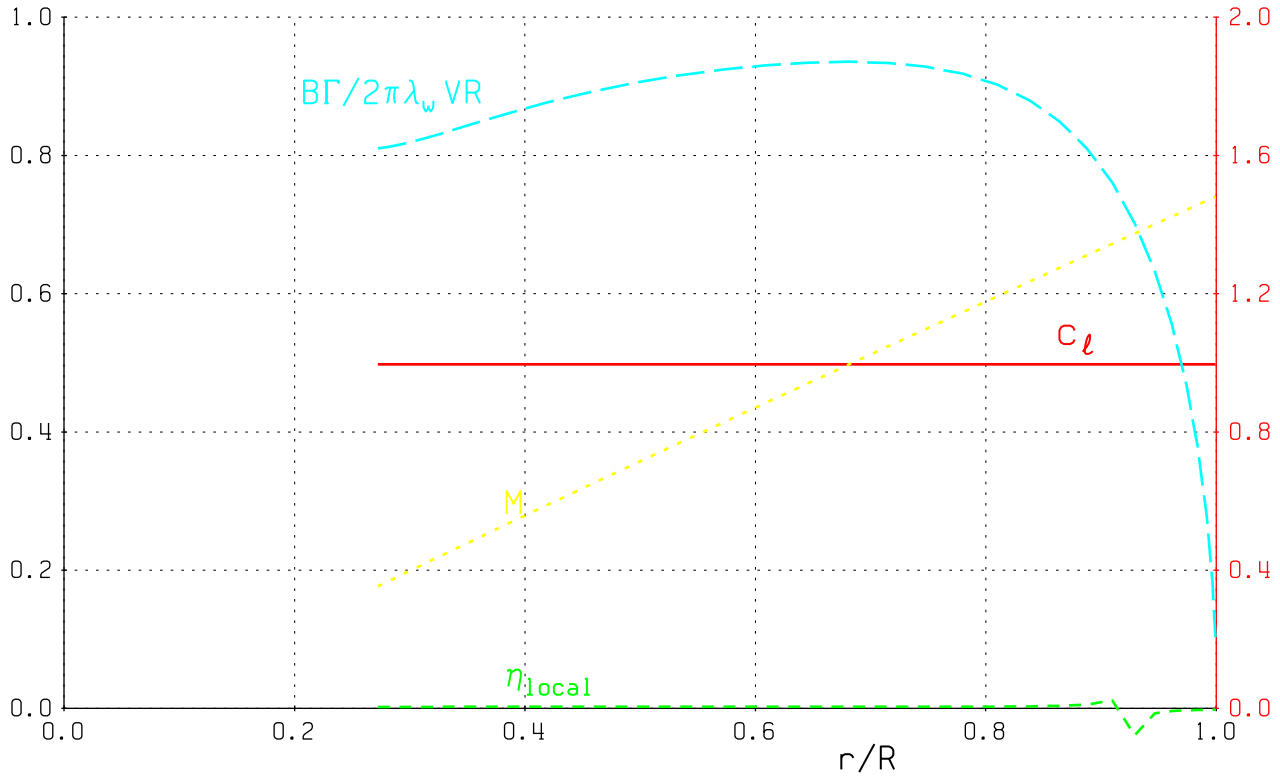
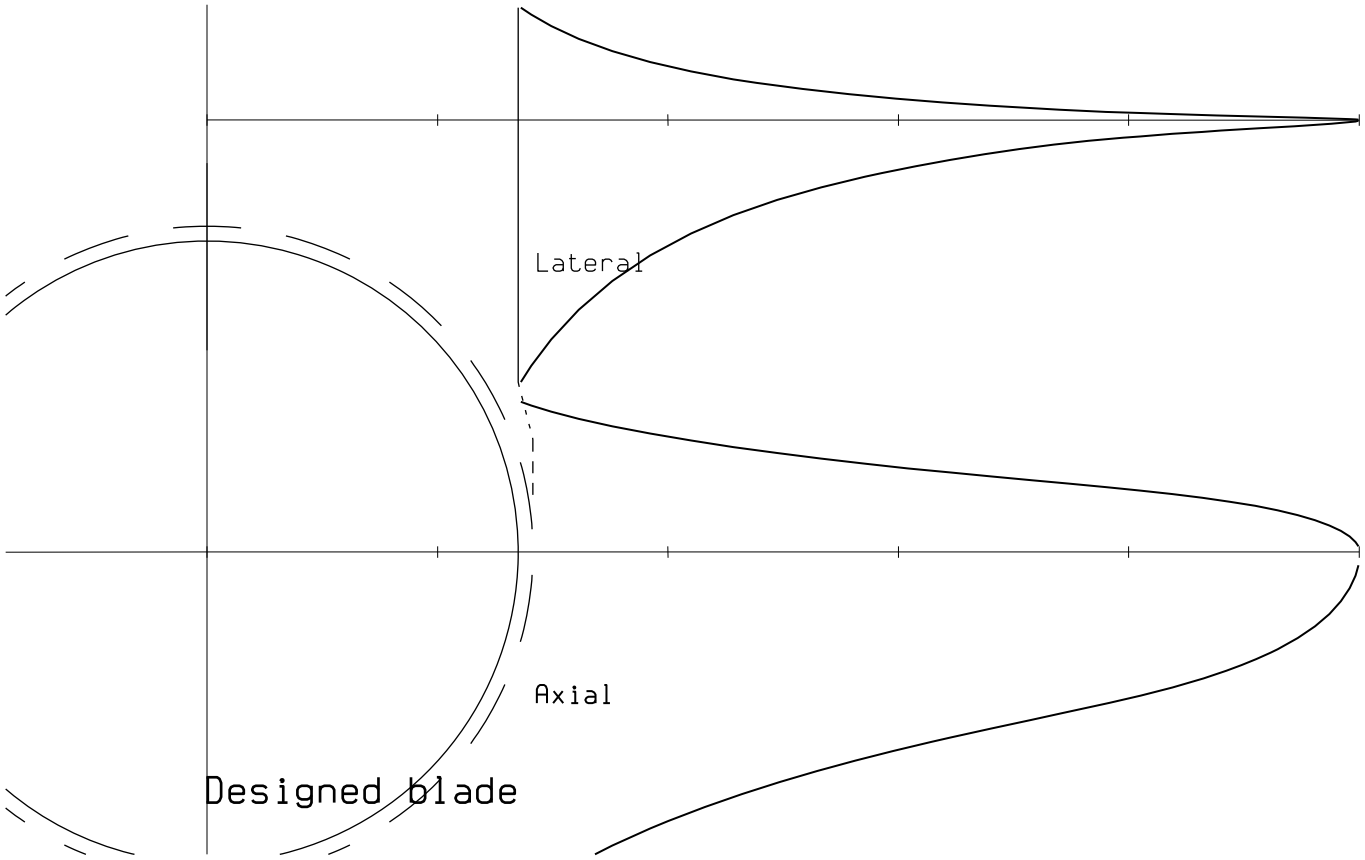
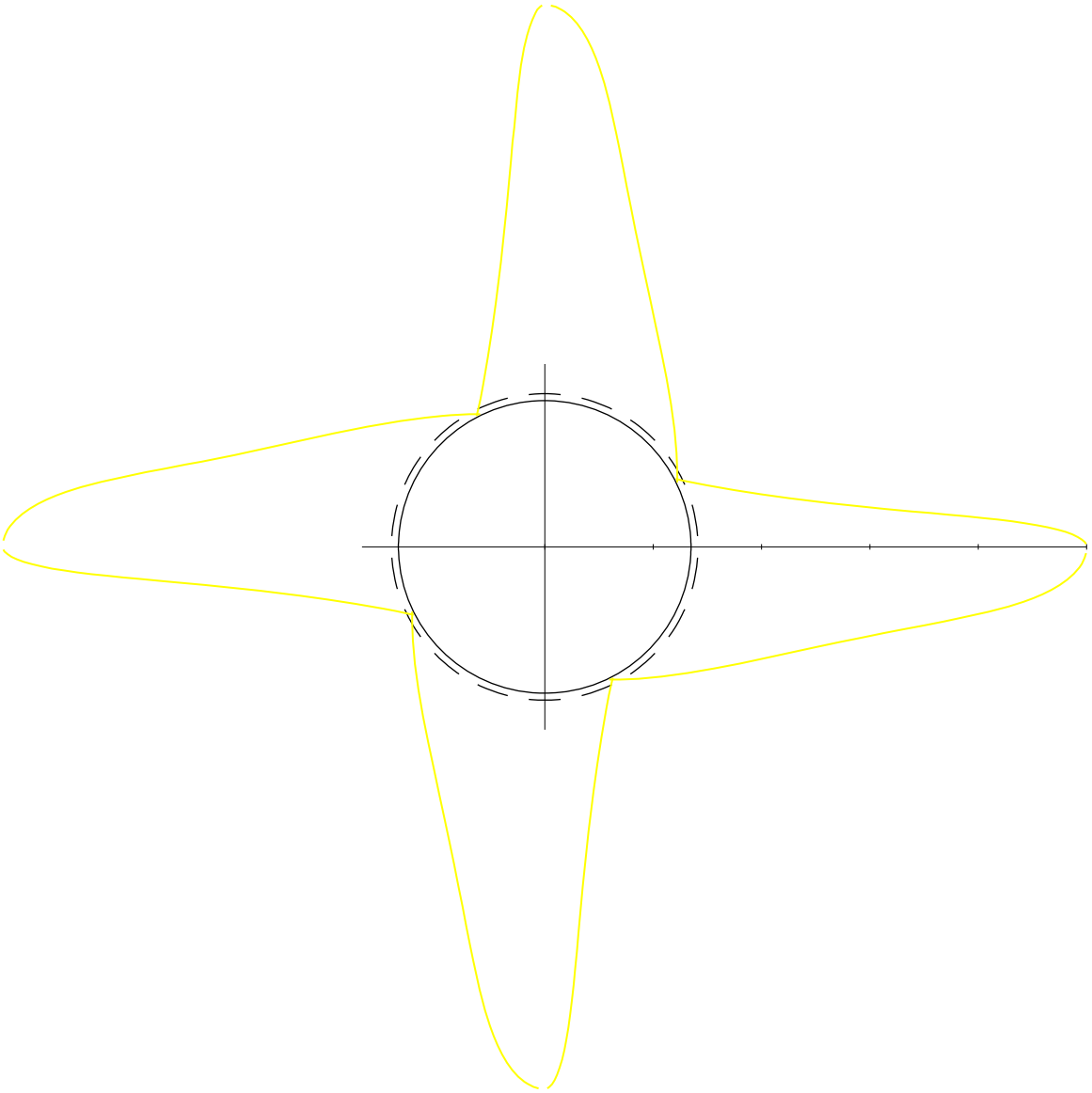
Designed blade

#bld= 4	R m = 0.780	$\sigma_{3/4}$ = 0.2541	β_{twist}= 33.556	
Vm/s= 0.100	V/ΩR= 0.0004	P_C =	C_P = 0.1048	η_{ideal}= 0.0029
h km= 0.000	J = 0.0012	T_C =	C_T = 0.2807	η = 0.0033
T kN= 5.5181	P kW= 167.1828	RPM = 3121.9	β_{tip} = 3.215	
Helicopter	C_{TH} = 0.036214	C_{PH} = 0.004303	C_{TH}/σ= 0.1425	FOM = 1.1326





Designed blade

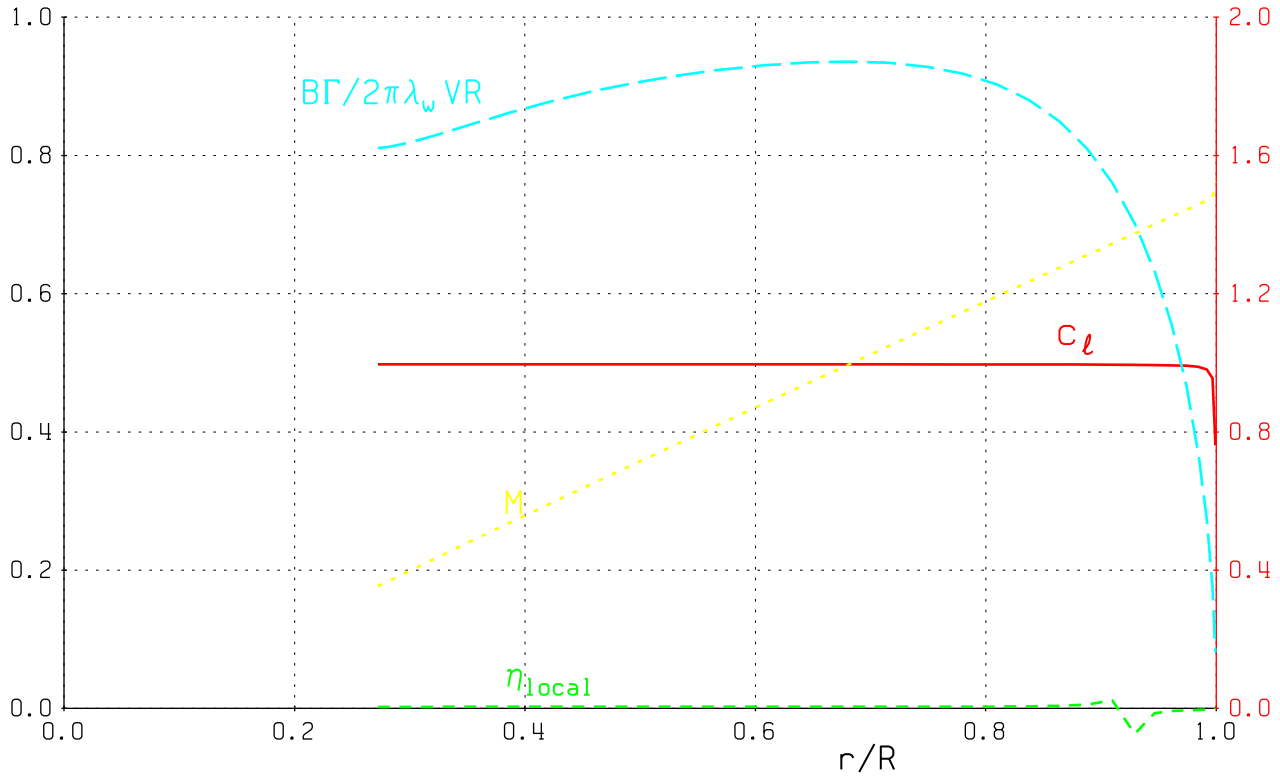


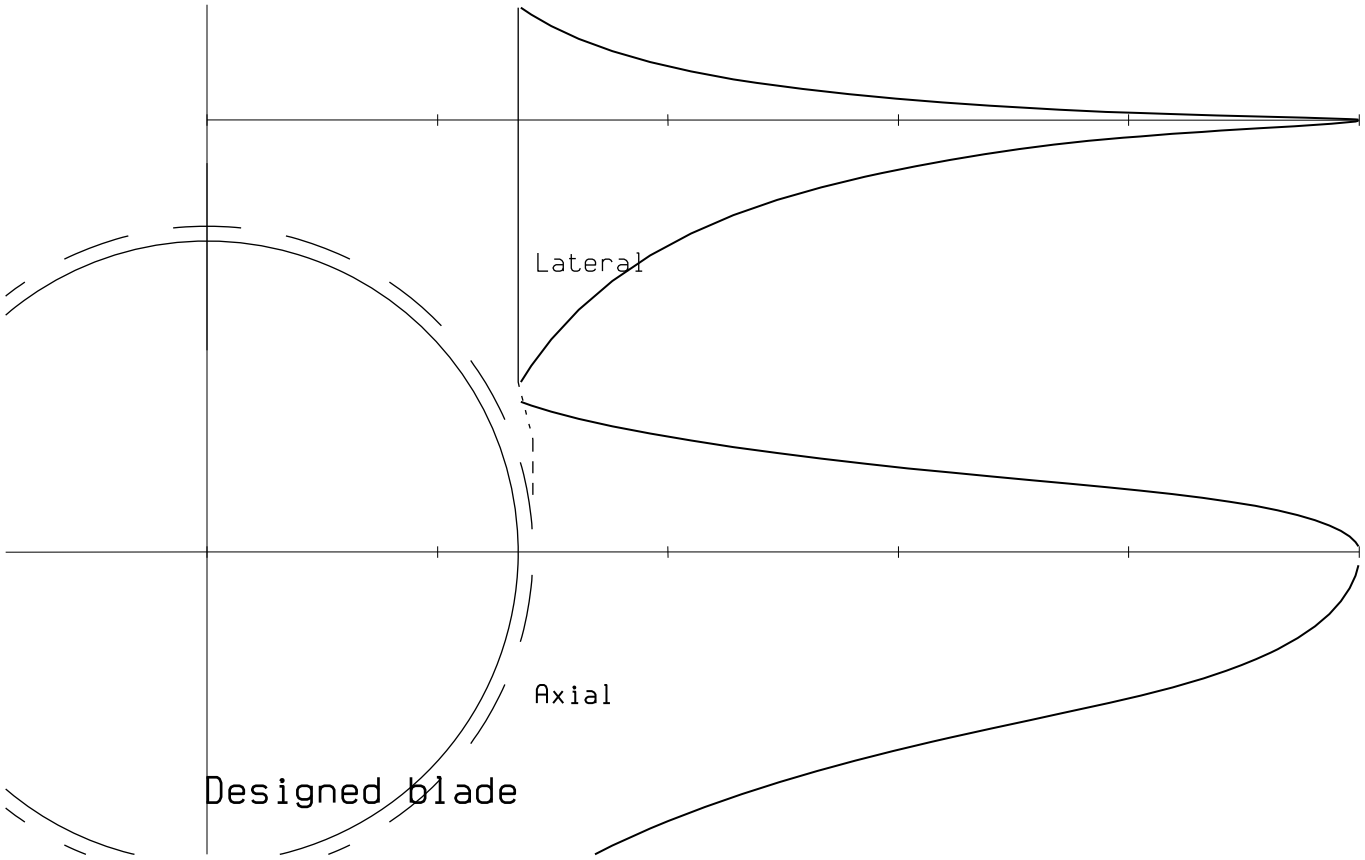
Designed blade

#bld	= 4	R m	= 0.7800	A m²	= 1.772008
$\sigma_{3/4}$	= 0.2541	R_{hub}	= 0.2106		
β_{twist}	= 33.556	R_{wak}	= 0.2206		

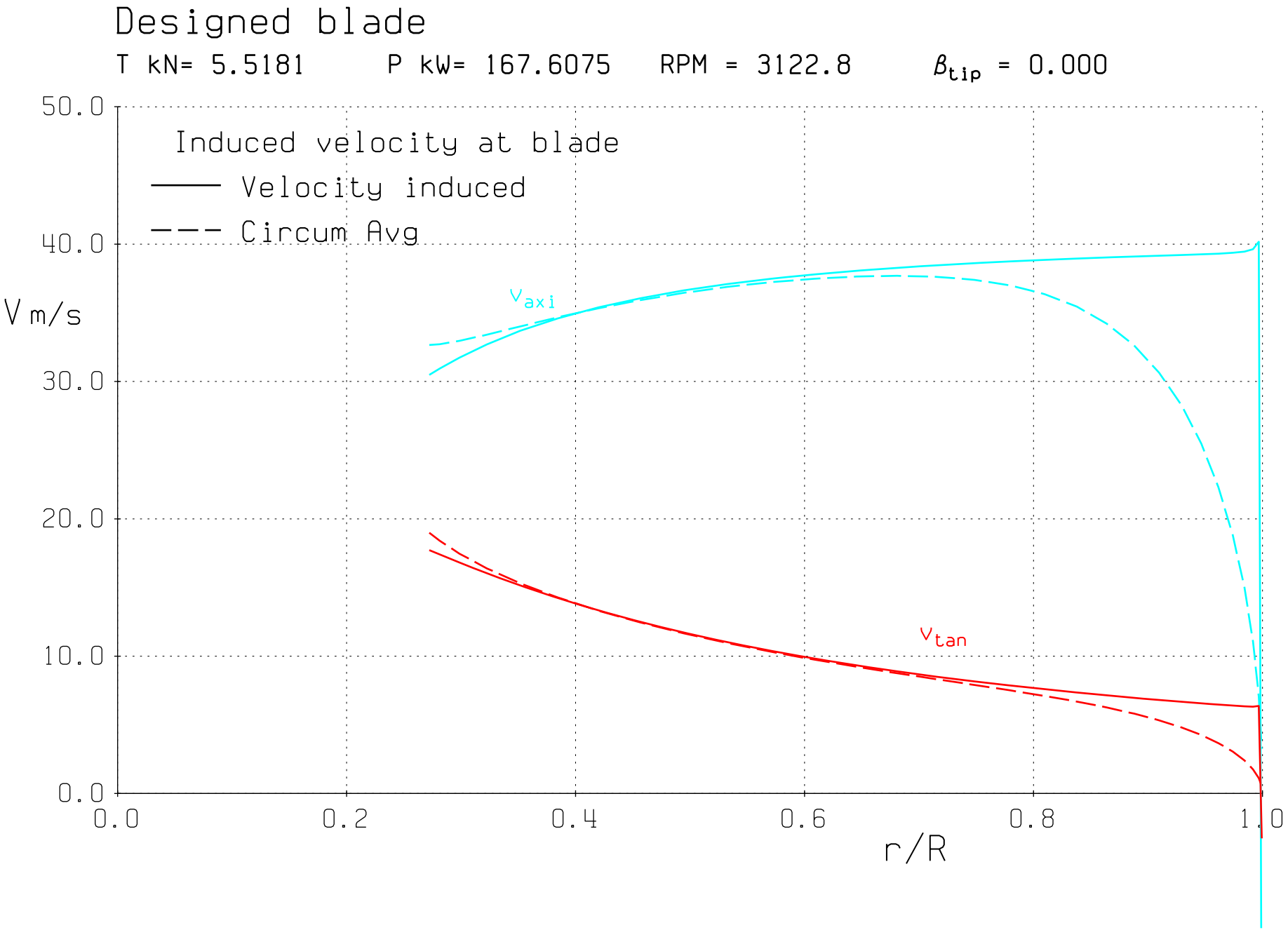
Designed blade

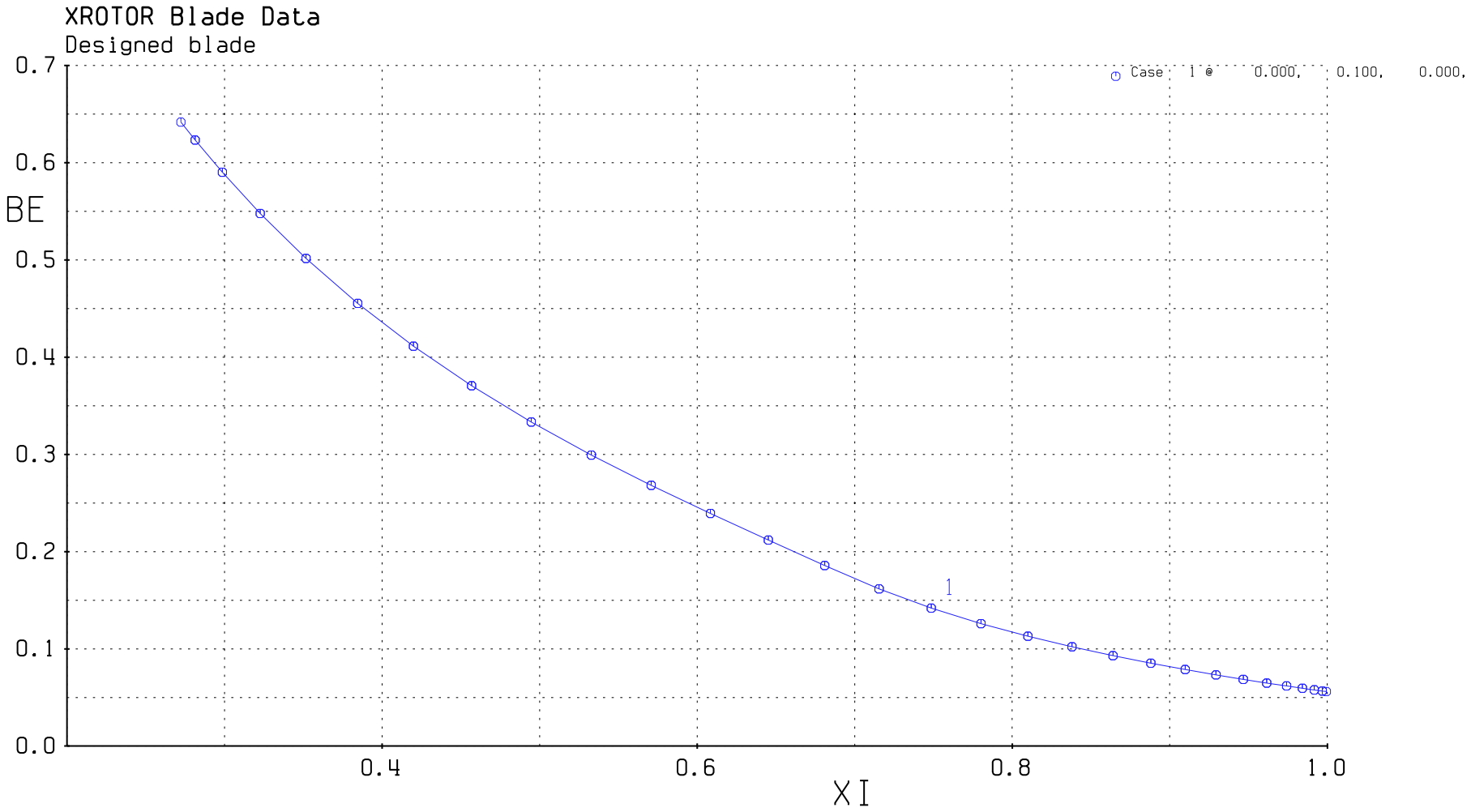
#bld= 4	R m = 0.780	$\sigma_{3/4}$ = 0.2541	β_{twist}= 33.556	
Vm/s= 0.100	V/ΩR= 0.0004	P_C =	C_P = 0.1050	η_{ideal}= 0.0029
h km= 0.000	J = 0.0012	T_C =	C_T = 0.2806	η = 0.0033
T kN= 5.5181	P kW= 167.6075	RPM = 3122.8	β_{tip} = 3.215	
Helicopter	C_{TH} = 0.036194	C_{PH} = 0.004310	C_{TH}/σ= 0.1425	FOM = 1.1297

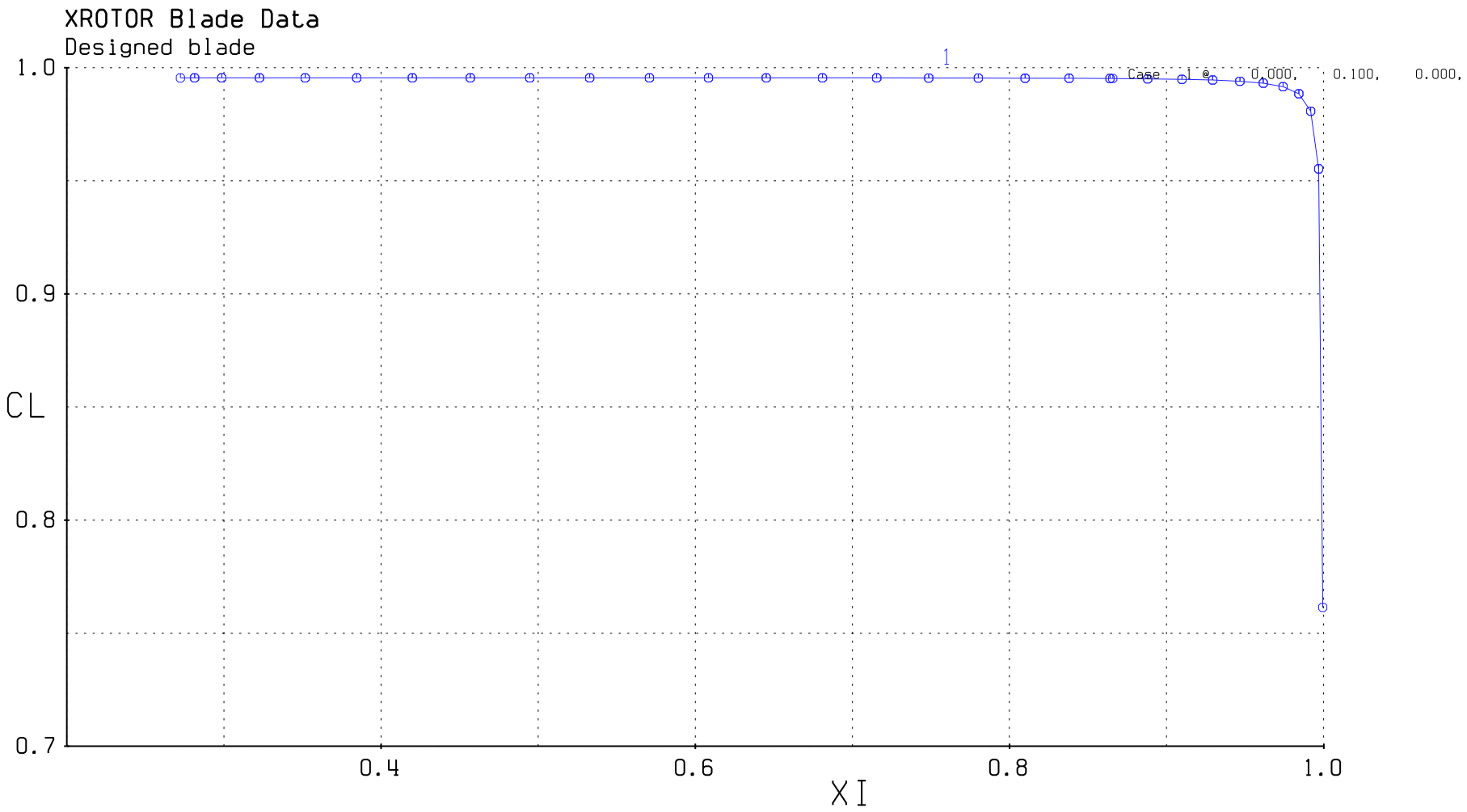


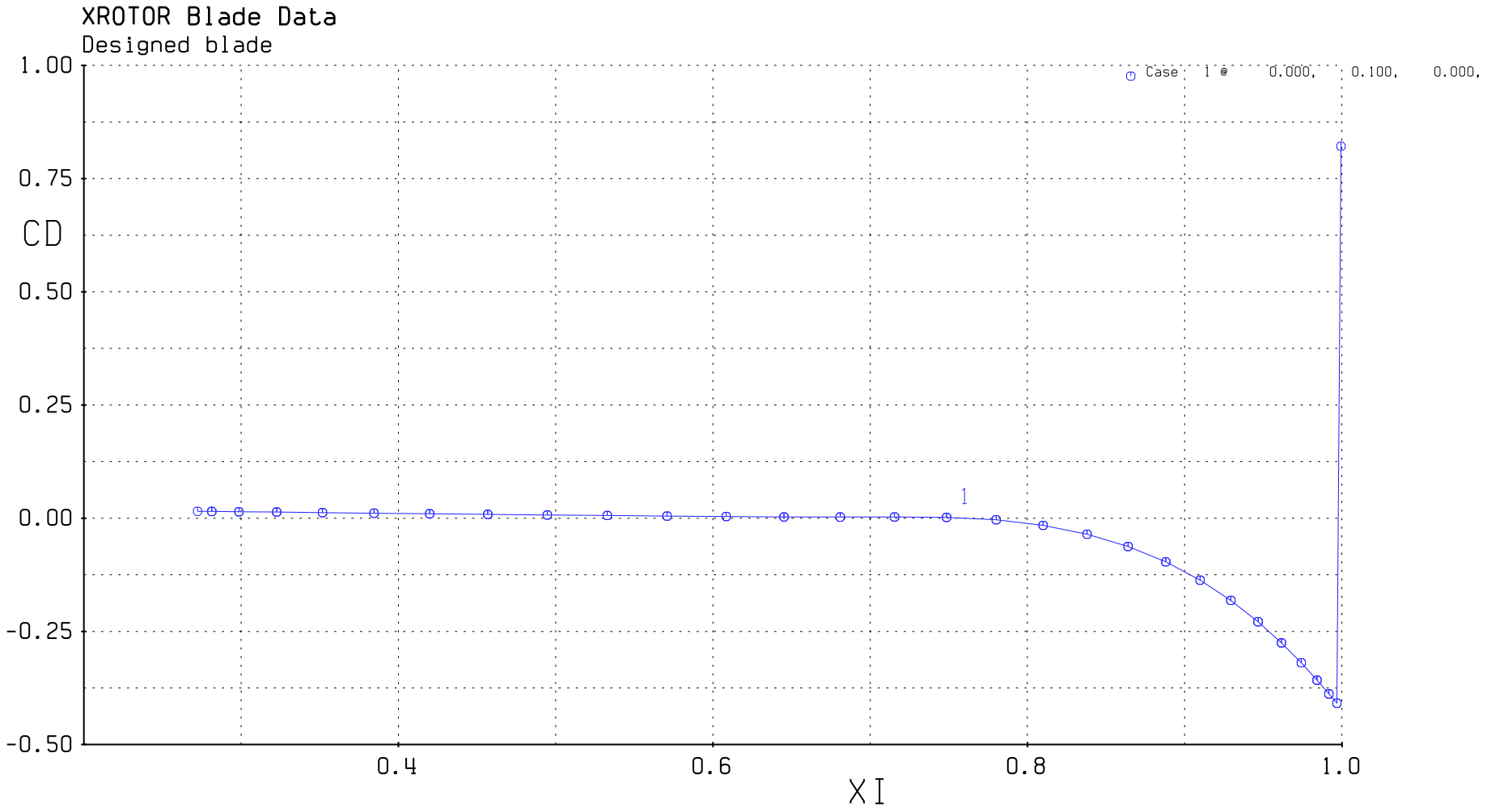


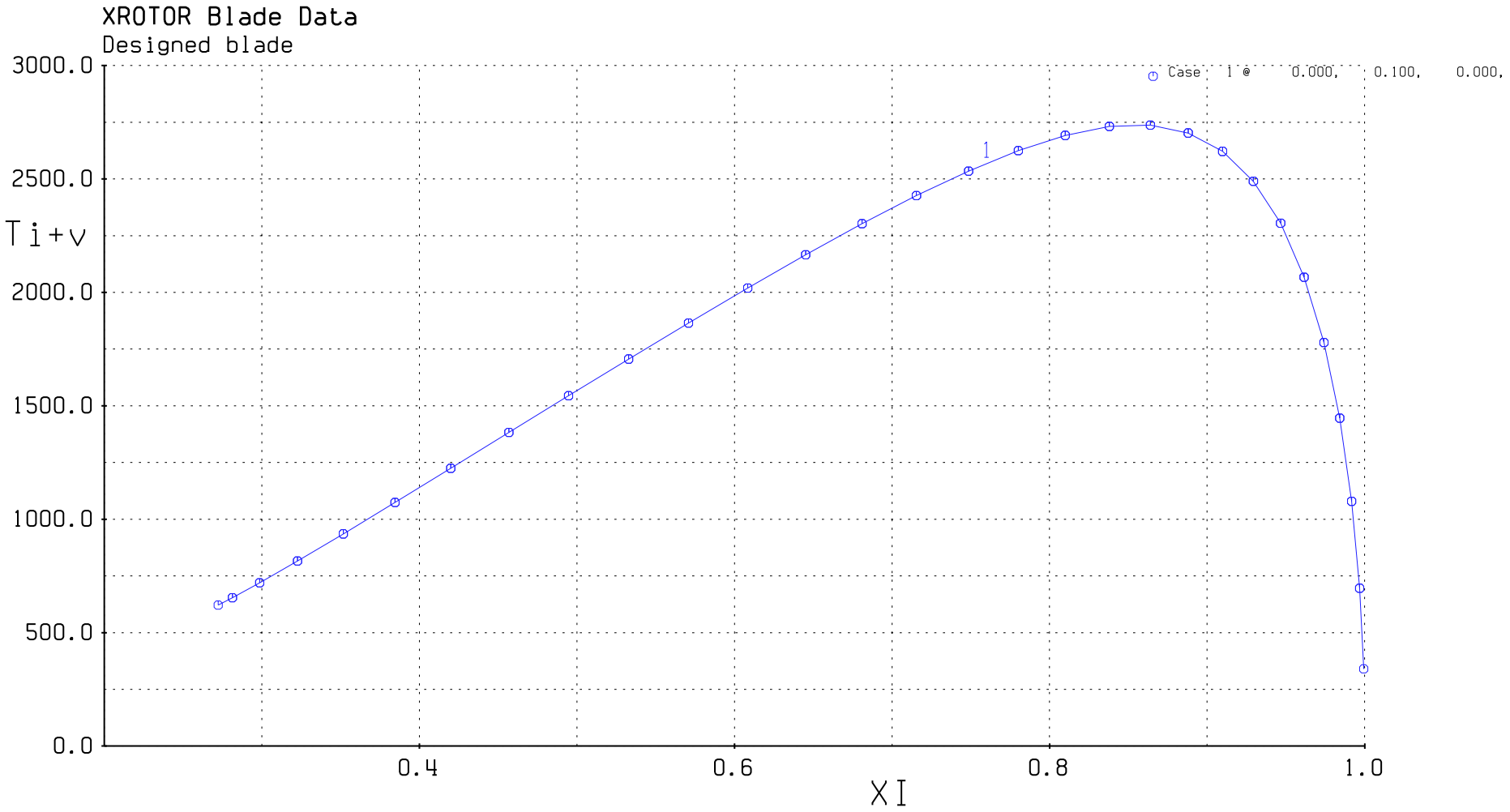
Designed blade

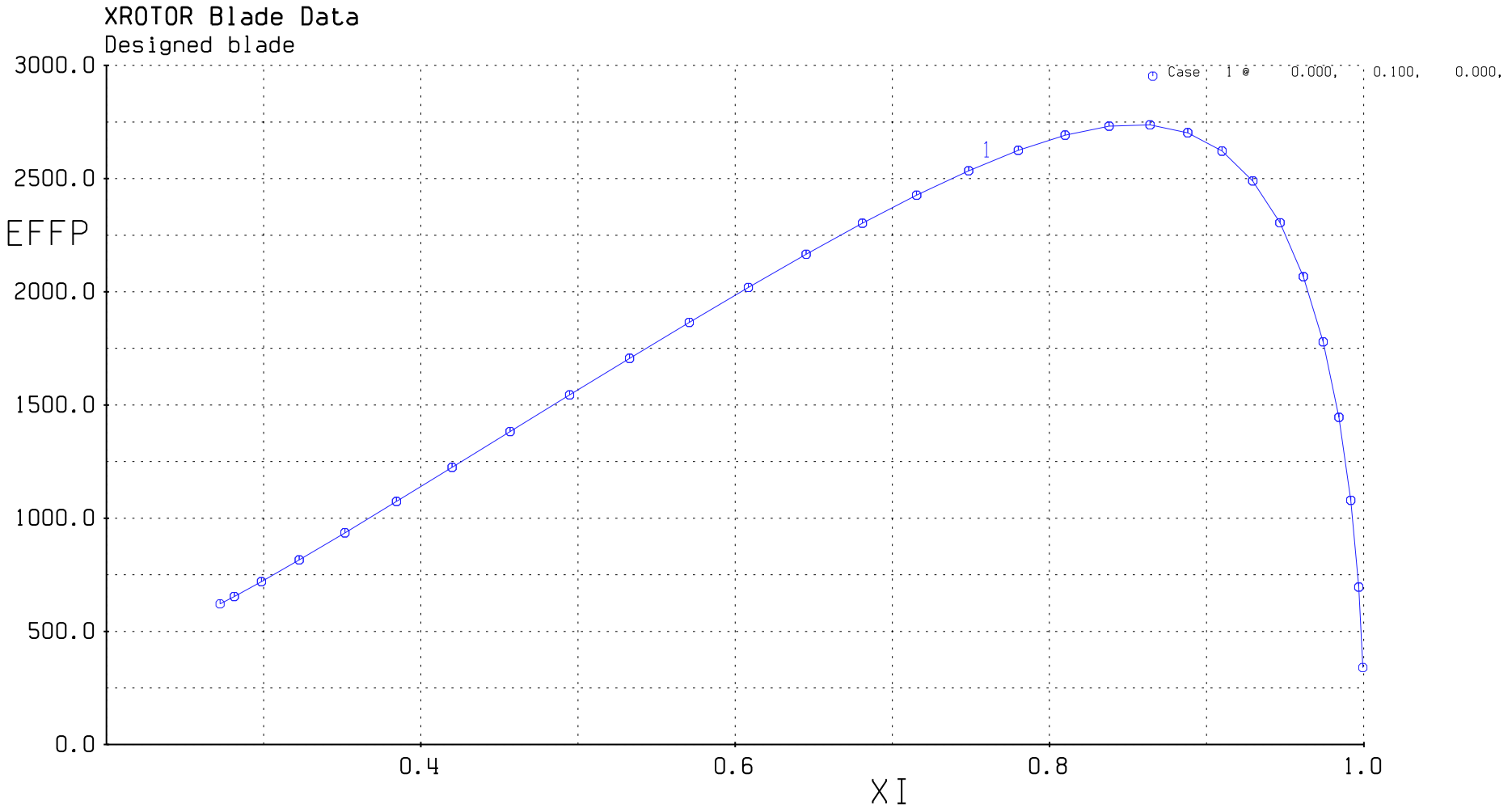

Designed blade
T kN= 5.5181
P kW= 167.6075
RPM = 3122.8
β_{tip} = 0.000
Induced velocity at blade
Velocity induced
Circum Avg
V_{axi}
V_{tan}
V m/s
r/R
50.0
40.0
30.0
20.0
10.0
0.0
0.0
0.2
0.4
0.6
0.8
1.0


XROTOR Blade Data
Designed blade
Case 1 @ 0.000, 0.100, 0.000,
BE
XI
0.0
0.1
0.2
0.3
0.4
0.5
0.6
0.7
0.4
0.6
0.8
1.0
1


XROTOR Blade Data
Designed blade
1
Case 1 e 0.000, 0.100, 0.000,
1.0
0.9
0.8
0.7
CL
0.4
0.6
0.8
1.0
XI


XROTOR Blade Data
Designed blade
Case 1 @ 0.000, 0.100, 0.000,
1.00
0.75
CD
0.50
0.25
0.00
-0.25
-0.50
0.4
0.6
0.8
1.0
XI
1


XROTOR Blade Data
Designed blade
Case 1 @ 0.000, 0.100, 0.000,
3000.0
2500.0
2000.0
1500.0
1000.0
500.0
0.0
Ti+v
0.4
0.6
0.8
1.0
XI
1


XROTOR Blade Data
Designed blade
Case 1 @ 0.000, 0.100, 0.000,
3000.0
2500.0
2000.0
1500.0
1000.0
500.0
0.0
EFFP
0.4
0.6
0.8
1.0
XI
1