End

Untwisted

Lateral

Axial
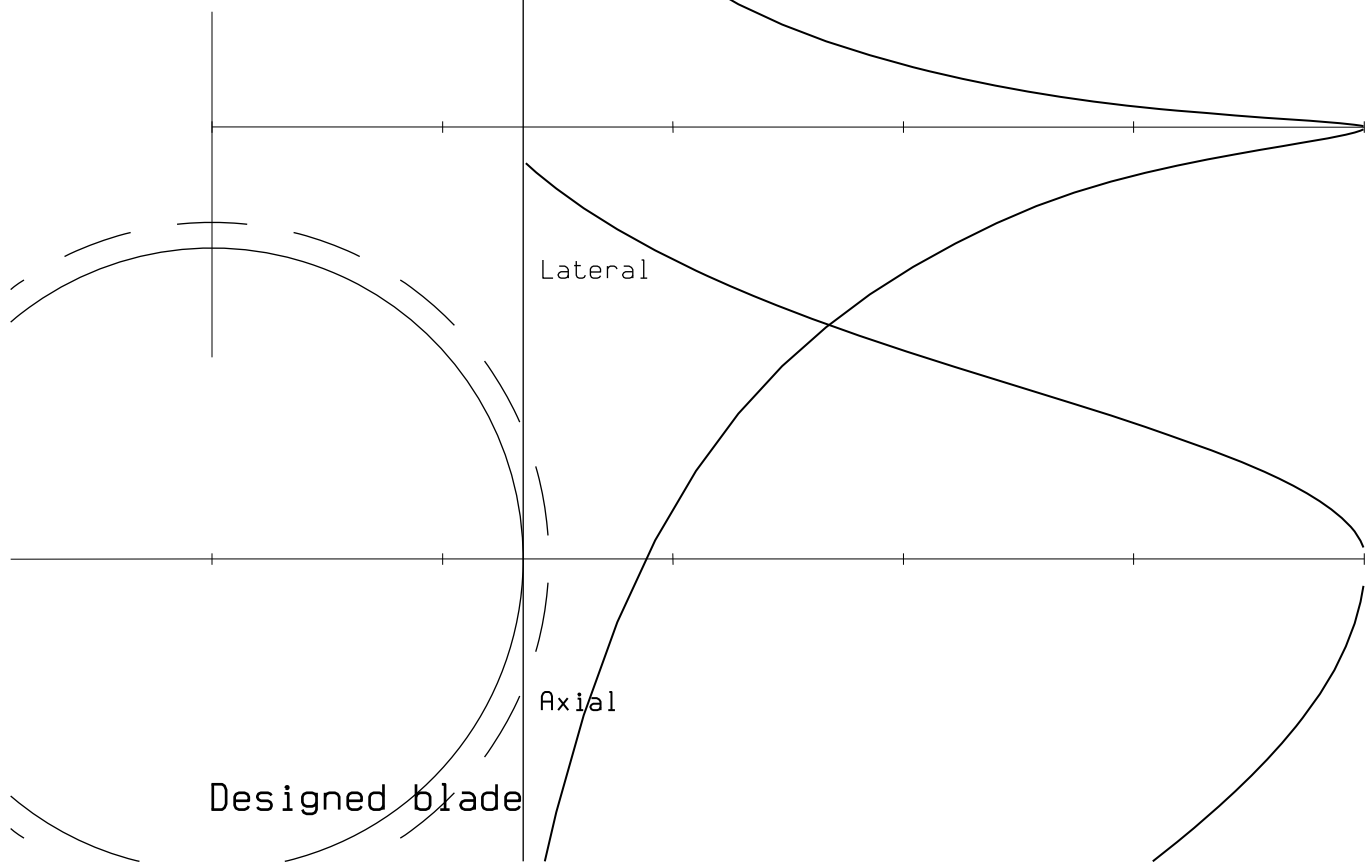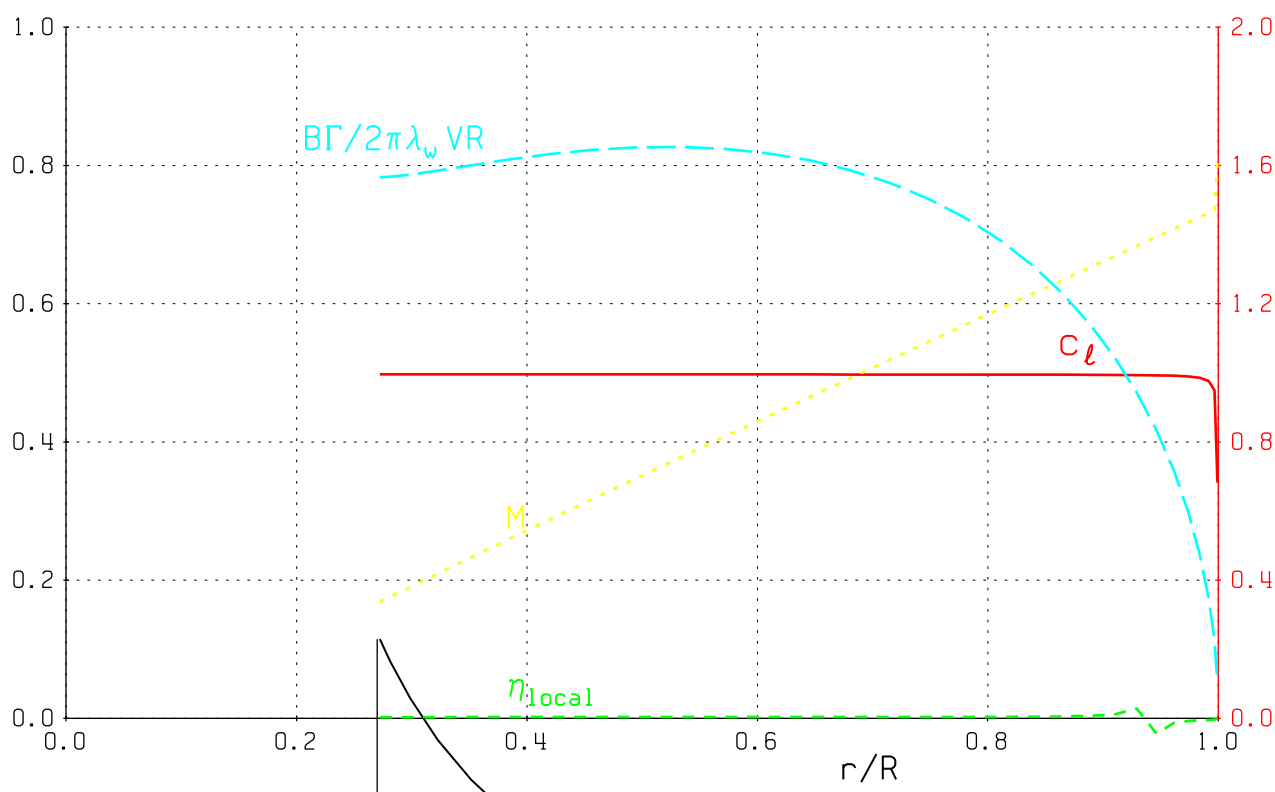
Designed blade

Designed blade

#bld = 2   R m   = 0.4500   A m² = 0.589796

$\sigma_{3/4}$ = 0.2893 R_{hub} = 0.1215
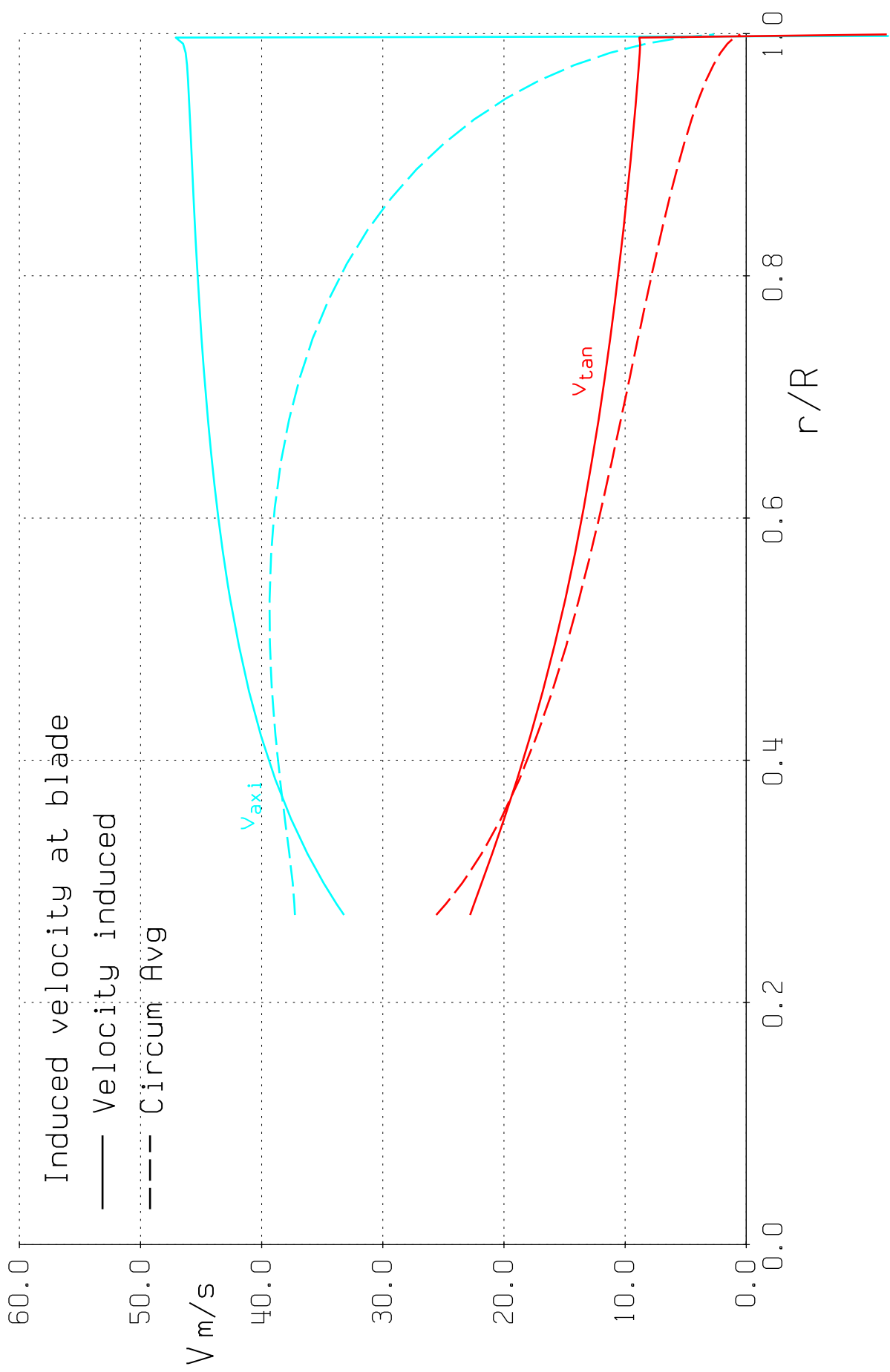
β_{twist} = 36.913 R_{wak} = 0.1315

Designed blade

#bld= 2	R m = 0.450	$\sigma_{3/4}$ = 0.2893	β_{twist}= 36.913		
Vm/s= 0.100	V/ΩR= 0.0004	P_C =	C_P = 0.1535	η_{ideal}= 0.0028	
h km= 0.000	J = 0.0012	T_C =	C_T = 0.3122	η = 0.0025	
T kN= 2.0438	P kW= 81.5985	RPM = 5413.1	β_{tip} = 4.818		

Helicopter C_{TH} = 0.040271 C_{PH} = 0.006303 C_{TH}/σ= 0.1392   FOM = 0.9066

$B\Gamma/2\pi\lambda_w VR$

C_ℓ

M

η_{local}

r/R

Lateral

Axial

Designed blade

Designed blade   P kW= 81.5985   RPM = 5413.1 β_{tip} = 0.000

T kN= 2.0438

Induced velocity at blade
—— Velocity induced
--- Circum Avg

V_{axi}

V_{tan}
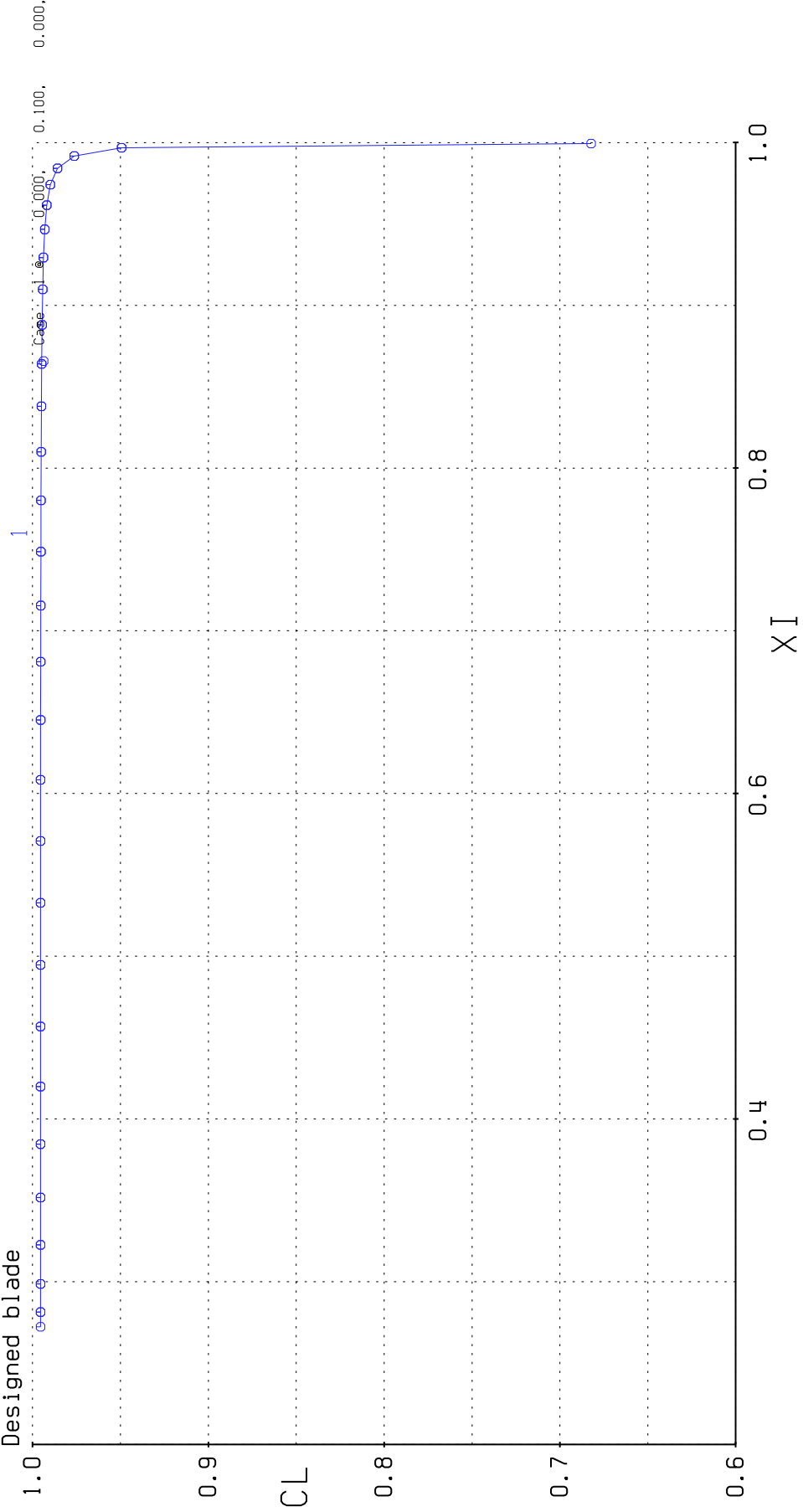
V m/s

r/R

XROTOR Blade Data
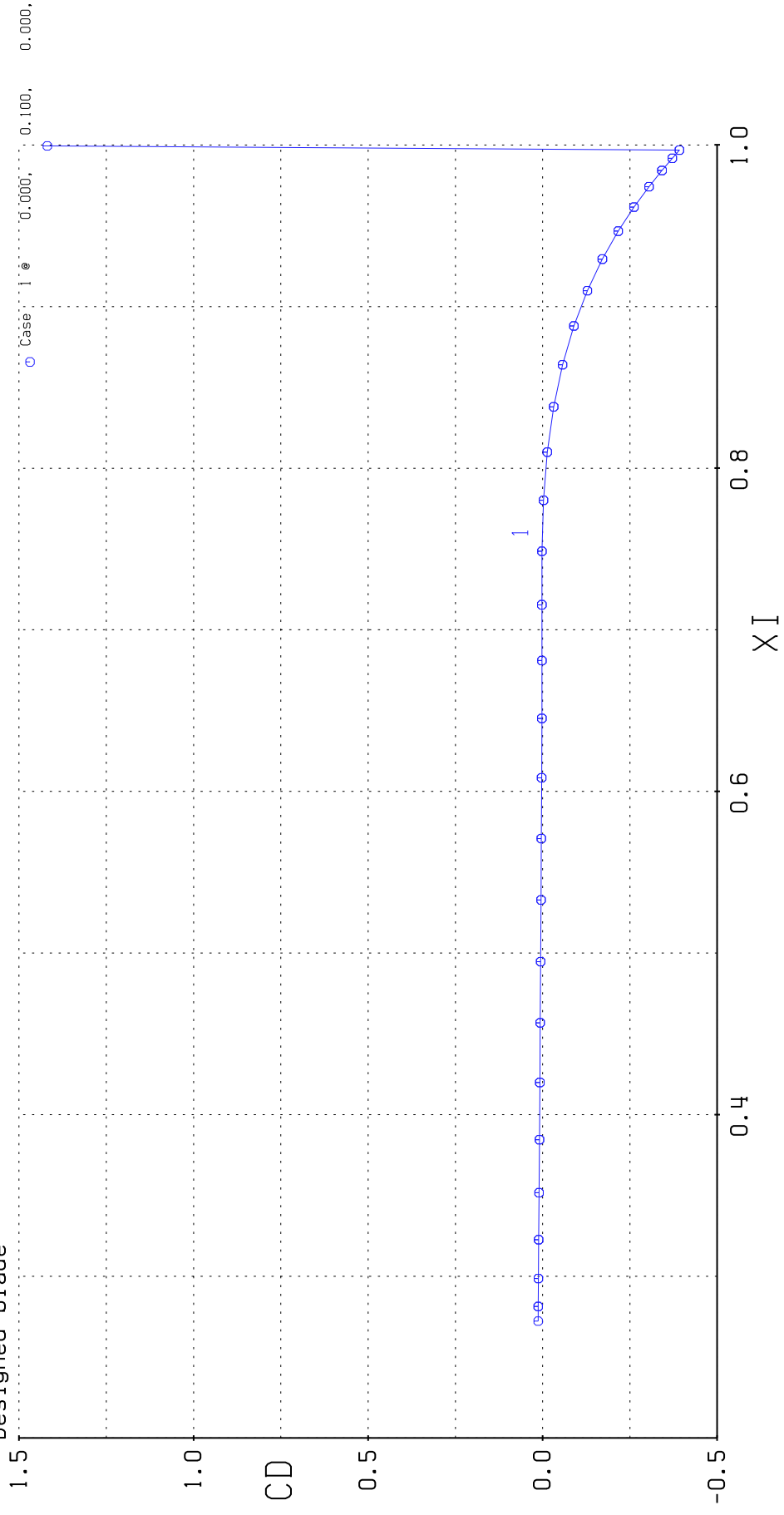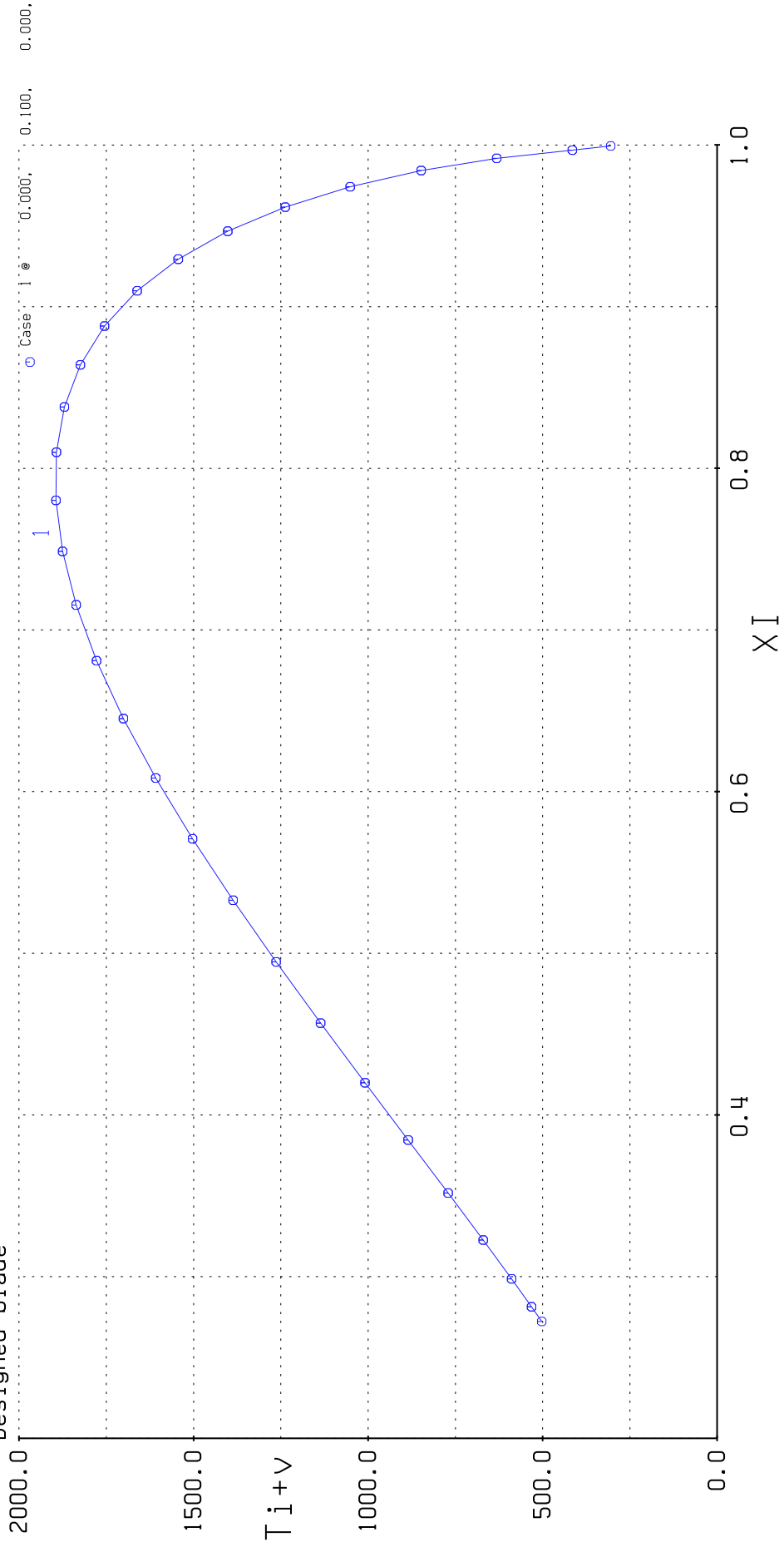Designed blade

Case 1

XI

BE

XROTOR Blade Data
Designed blade

XROTOR Blade Data
Designed blade

XROTOR Blade Data
Designed blade

XROTOR Blade Data
Designed blade

EFFP

XI

Case 1

2000.0   1500.0   1000.0   500.0   0.0

0.4   0.6   0.8   1.0

0.000.   0.100.   0.000.