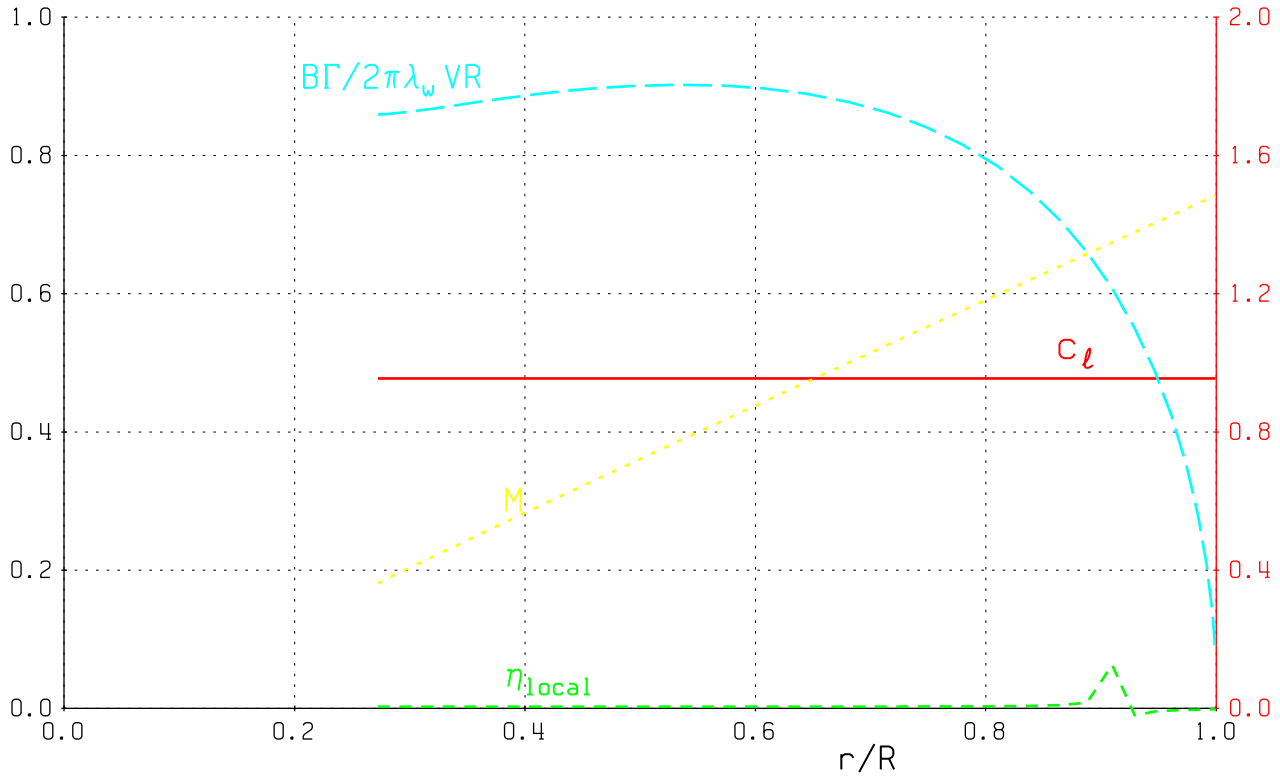
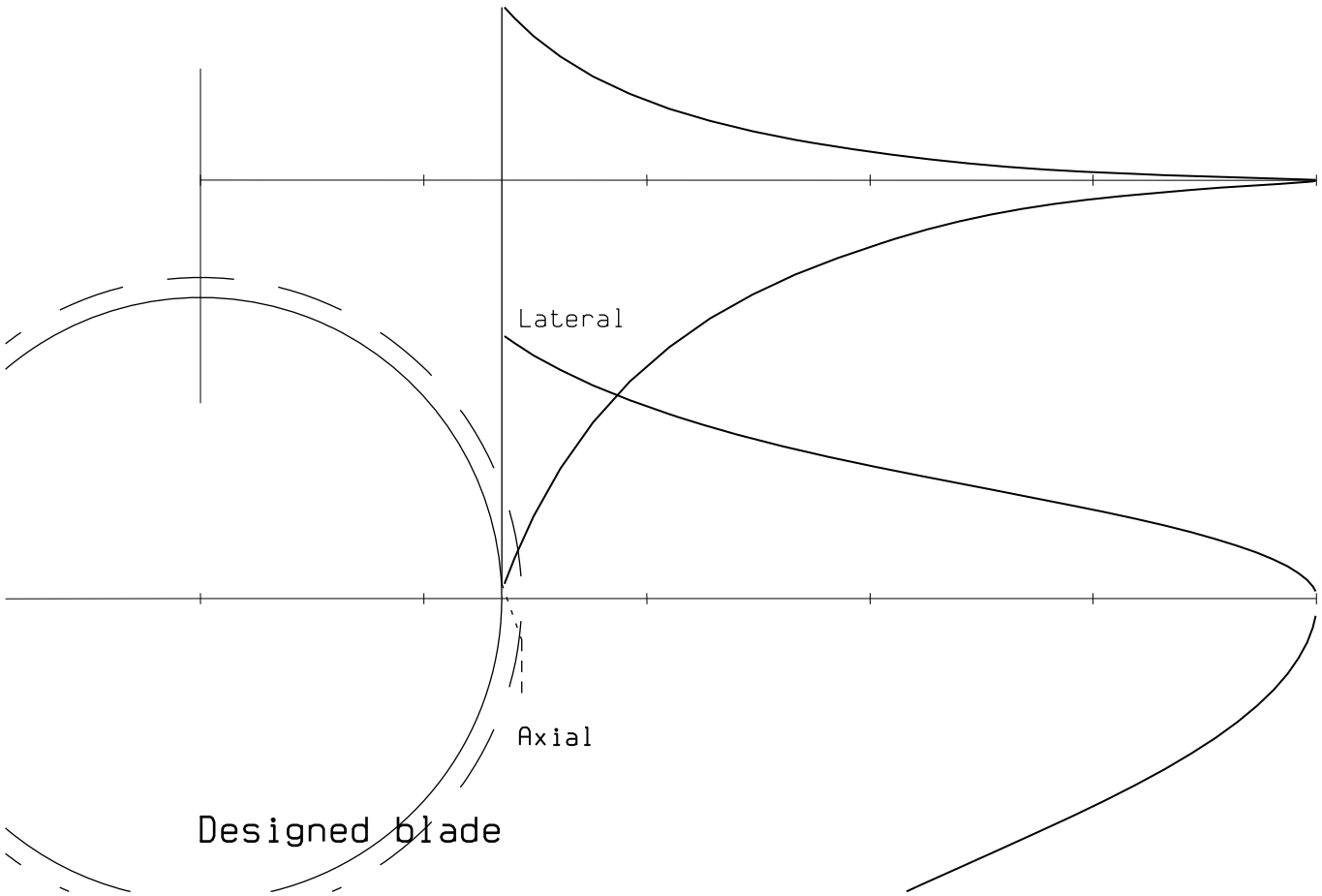
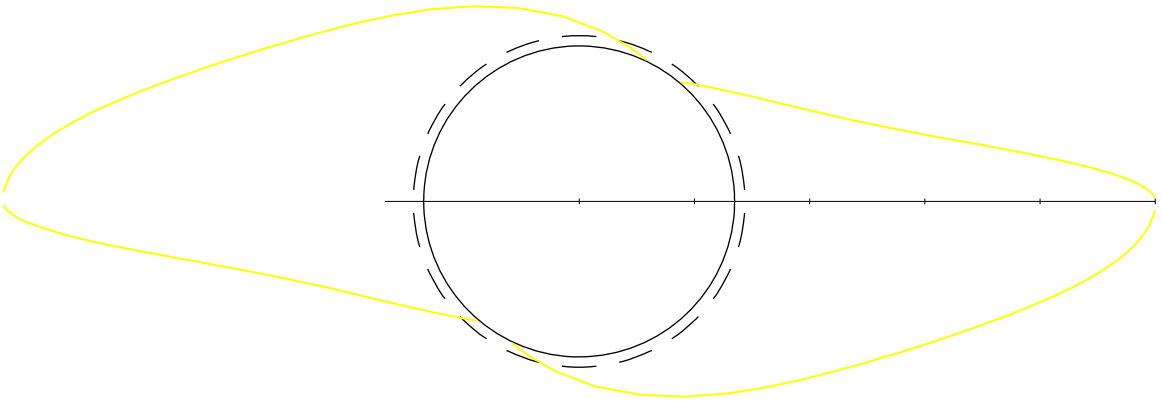
Designed blade

#bld= 2	R m = 0.565	$\sigma_{3/4}$ = 0.1916	β_{twist}= 31.249	
Vm/s= 0.100	V/ΩR= 0.0004	P_C =	C_P = 0.0687	η_{ideal}= 0.0034
h km= 0.000	J = 0.0012	T_C =	C_T = 0.2081	η = 0.0037
T kN= 2.1459	P kW= 57.5432	RPM = 4309.9	β_{tip} = 2.133	
Helicopter	C_{TH} = 0.026841	C_{PH} = 0.002823	C_{TH}/σ= 0.1401	FOM = 1.1017





Designed blade

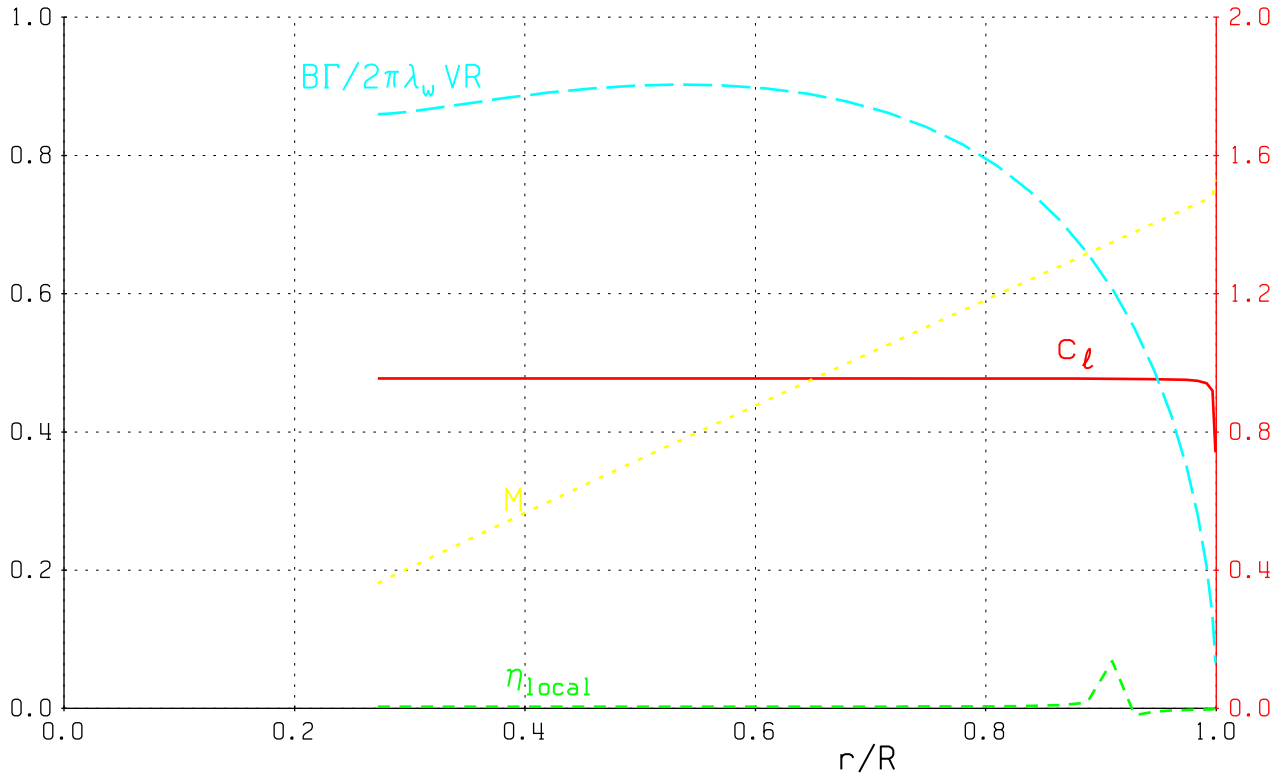


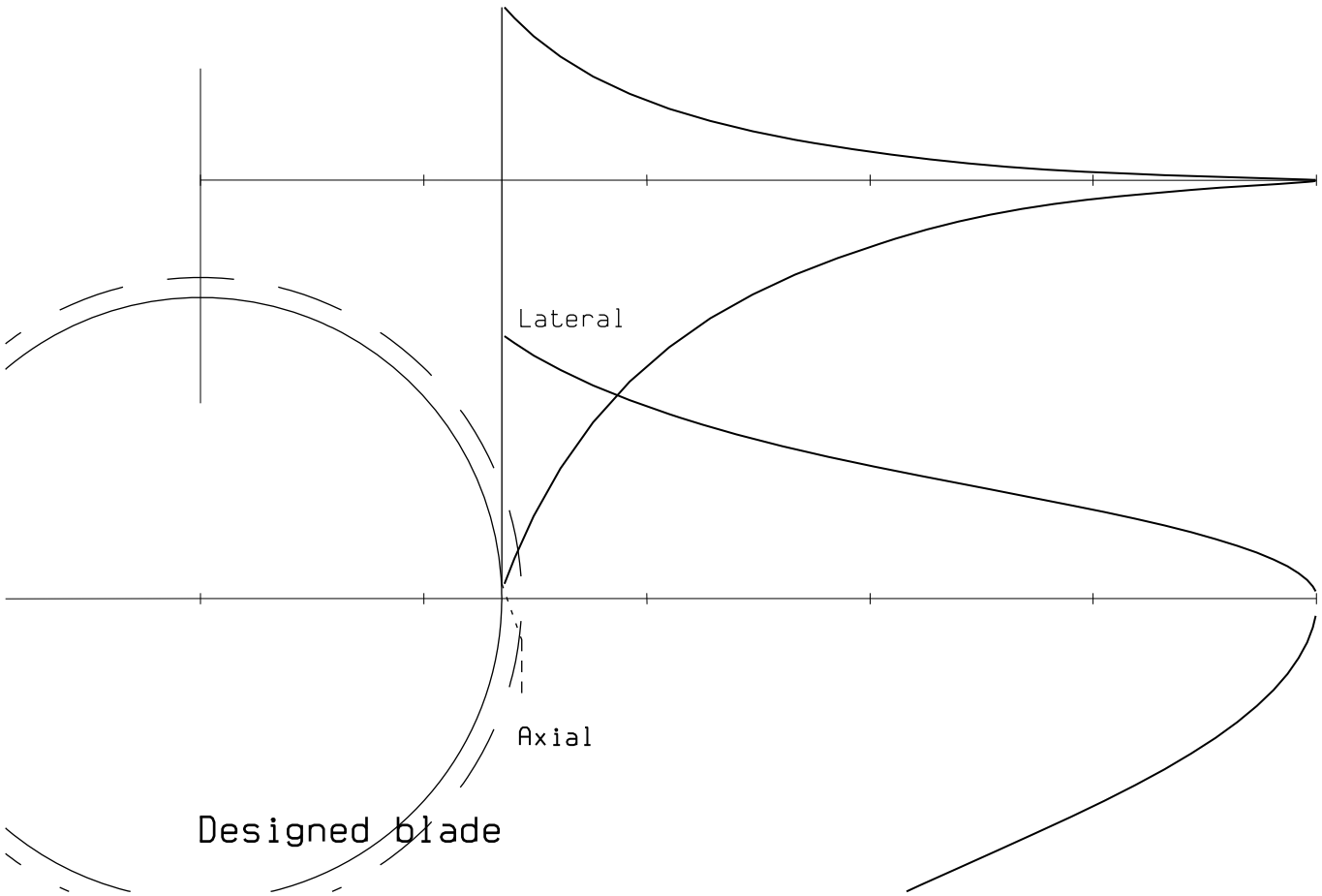
Designed blade

#bld	= 2	R m	= 0.5650	A m²	= 0.929813
$\sigma_{3/4}$	= 0.1916	R_{hub}	= 0.1525		
β_{twist}	= 31.249	R_{wak}	= 0.1626		

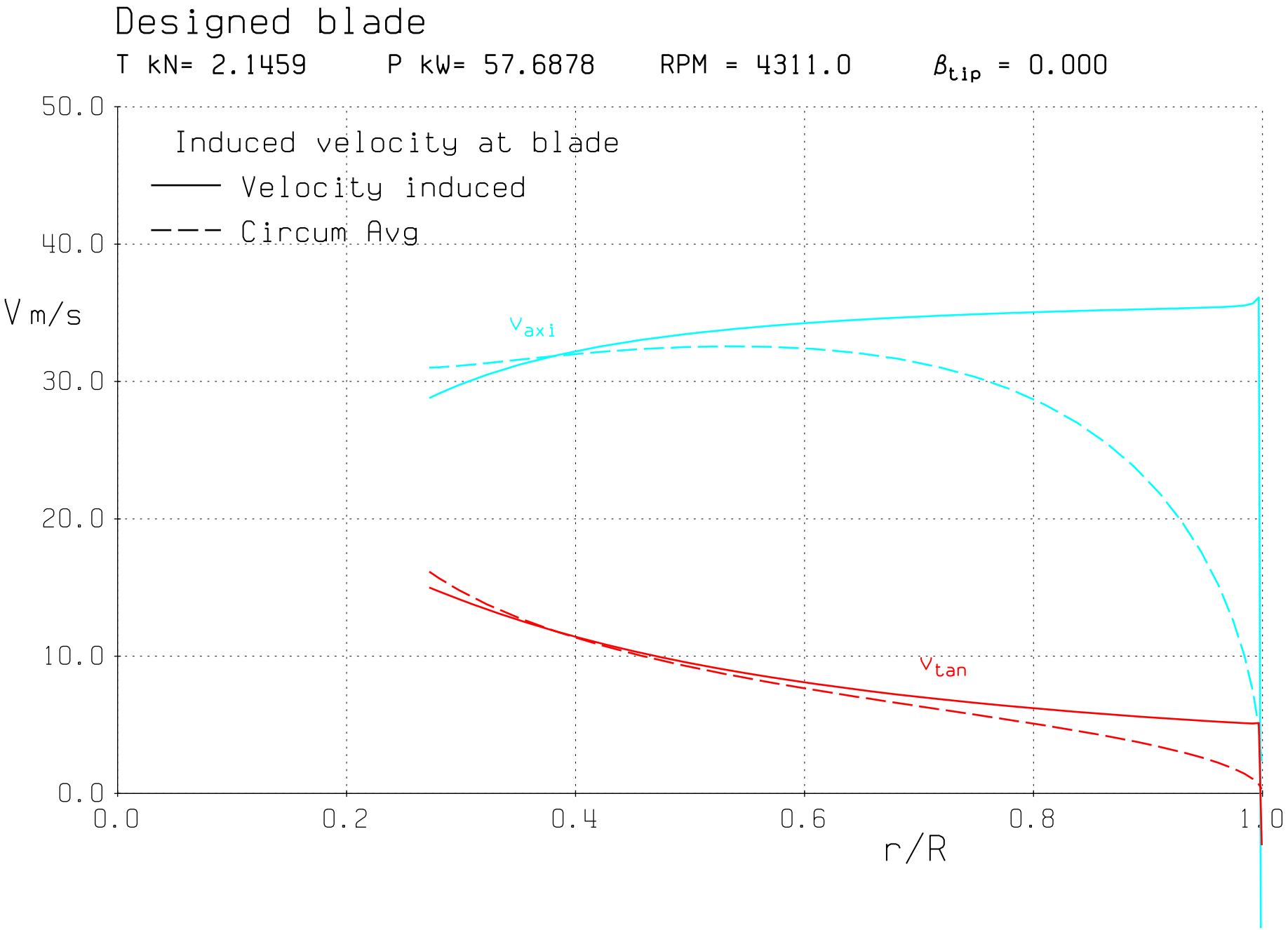
Designed blade

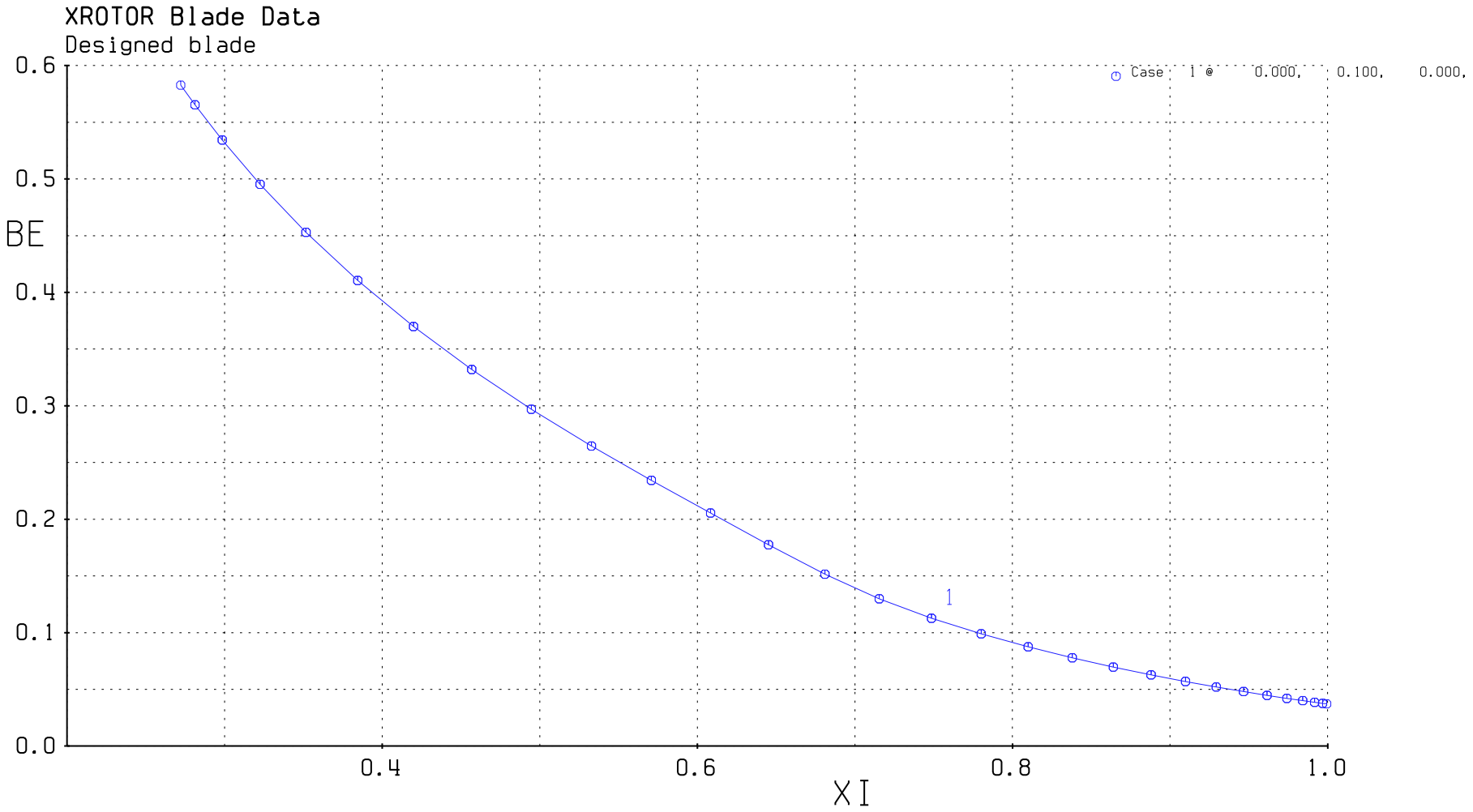
#bld= 2	R m = 0.565	$\sigma_{3/4}$ = 0.1916	β_{twist}= 31.249	
Vm/s= 0.100	V/ΩR= 0.0004	P_C =	C_P = 0.0689	η_{ideal}= 0.0034
h km= 0.000	J = 0.0012	T_C =	C_T = 0.2079	η = 0.0037
T kN= 2.1459	P kW= 57.6878	RPM = 4311.0	β_{tip} = 2.133	
Helicopter	C_{TH} = 0.026826	C_{PH} = 0.002827	C_{TH}/σ= 0.1400	FOM = 1.0989

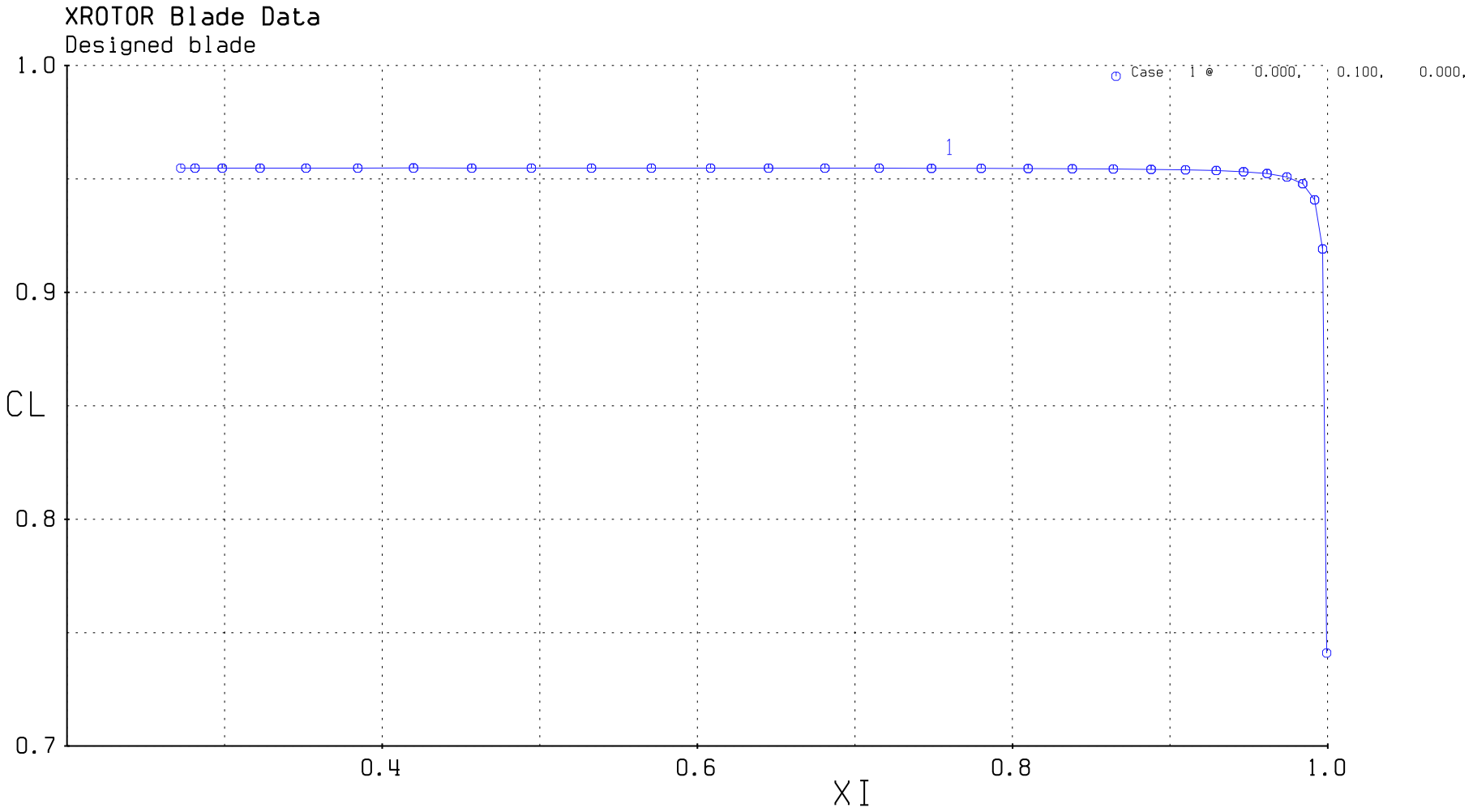


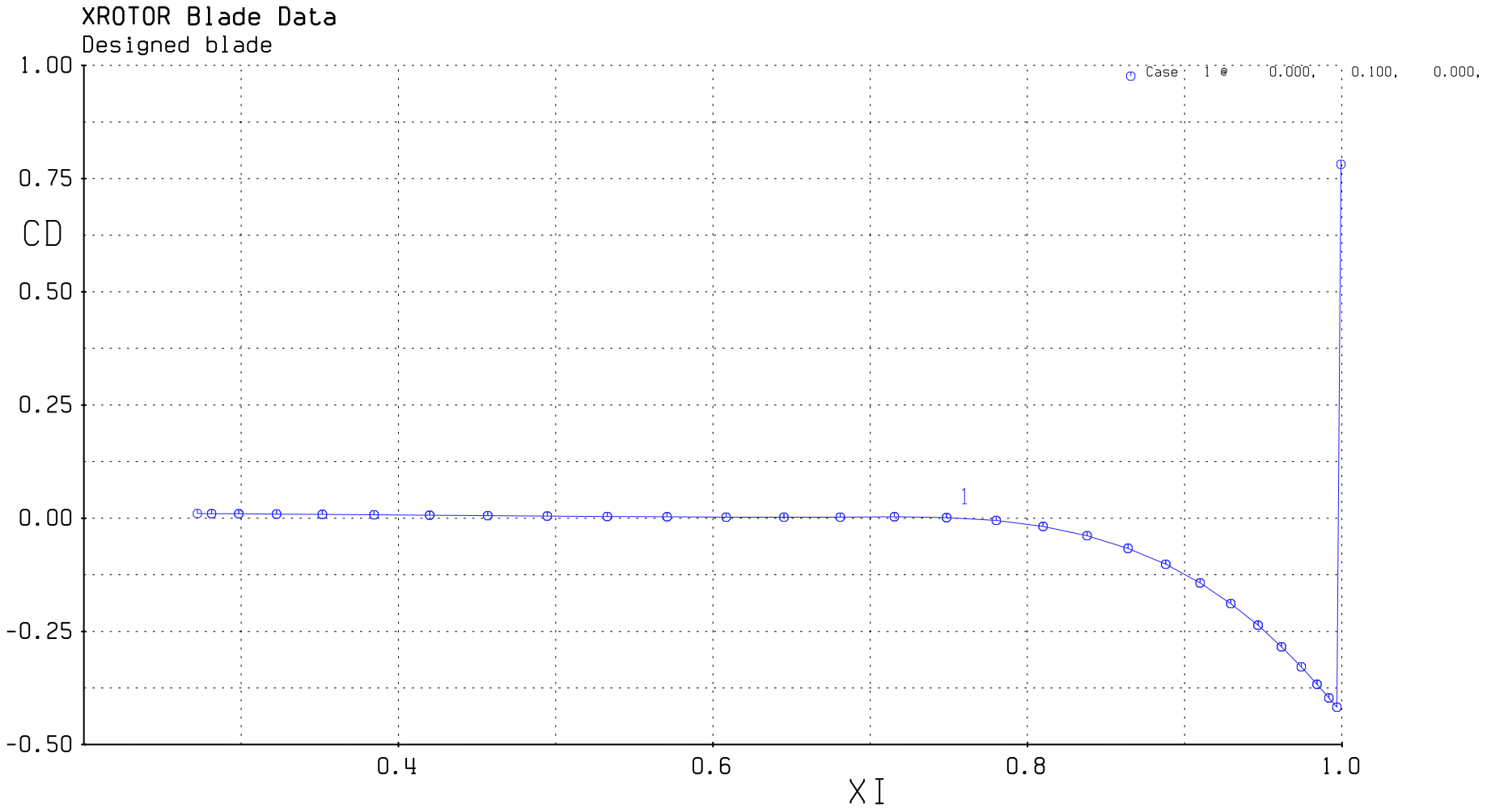


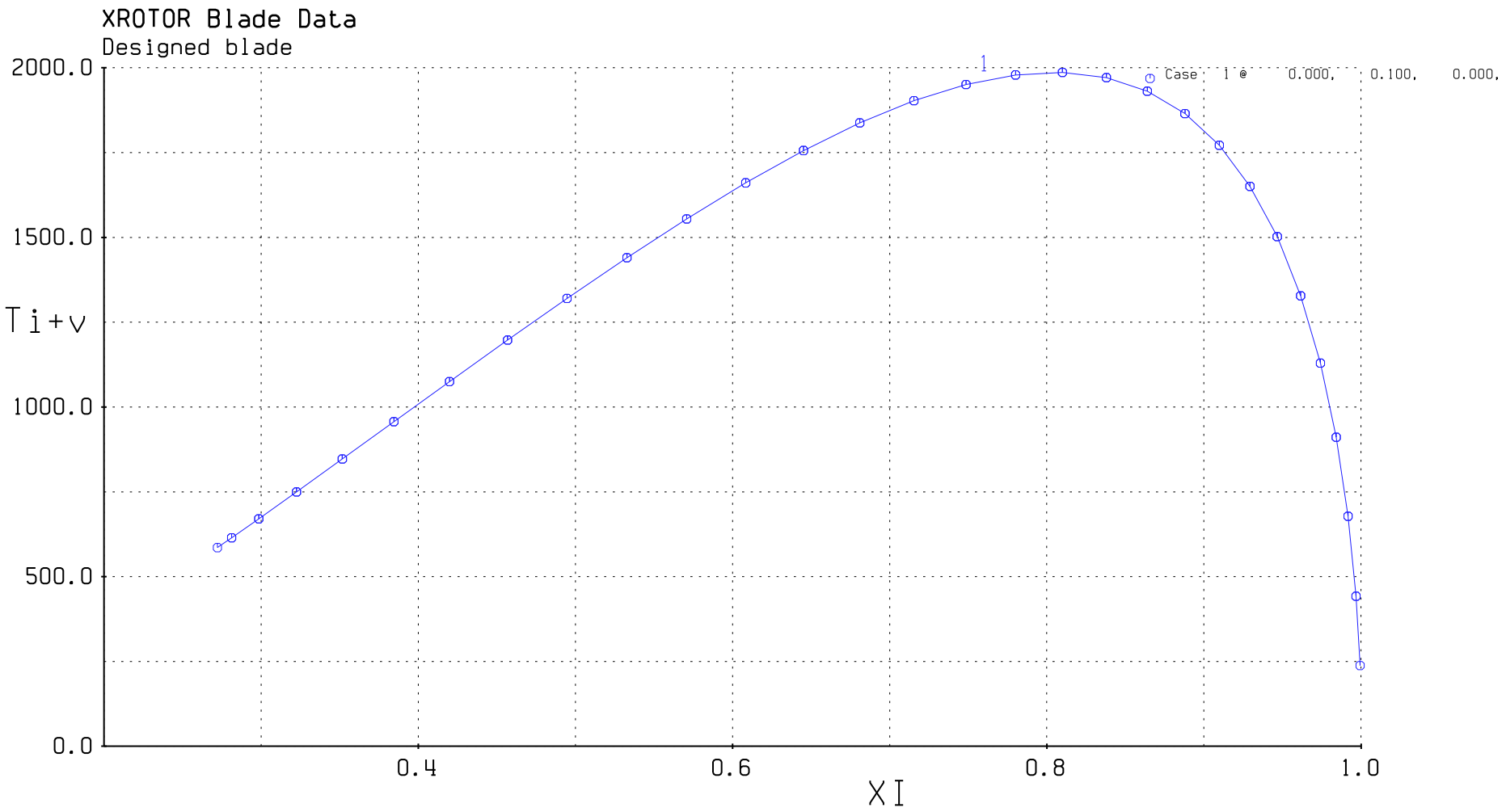
Designed blade

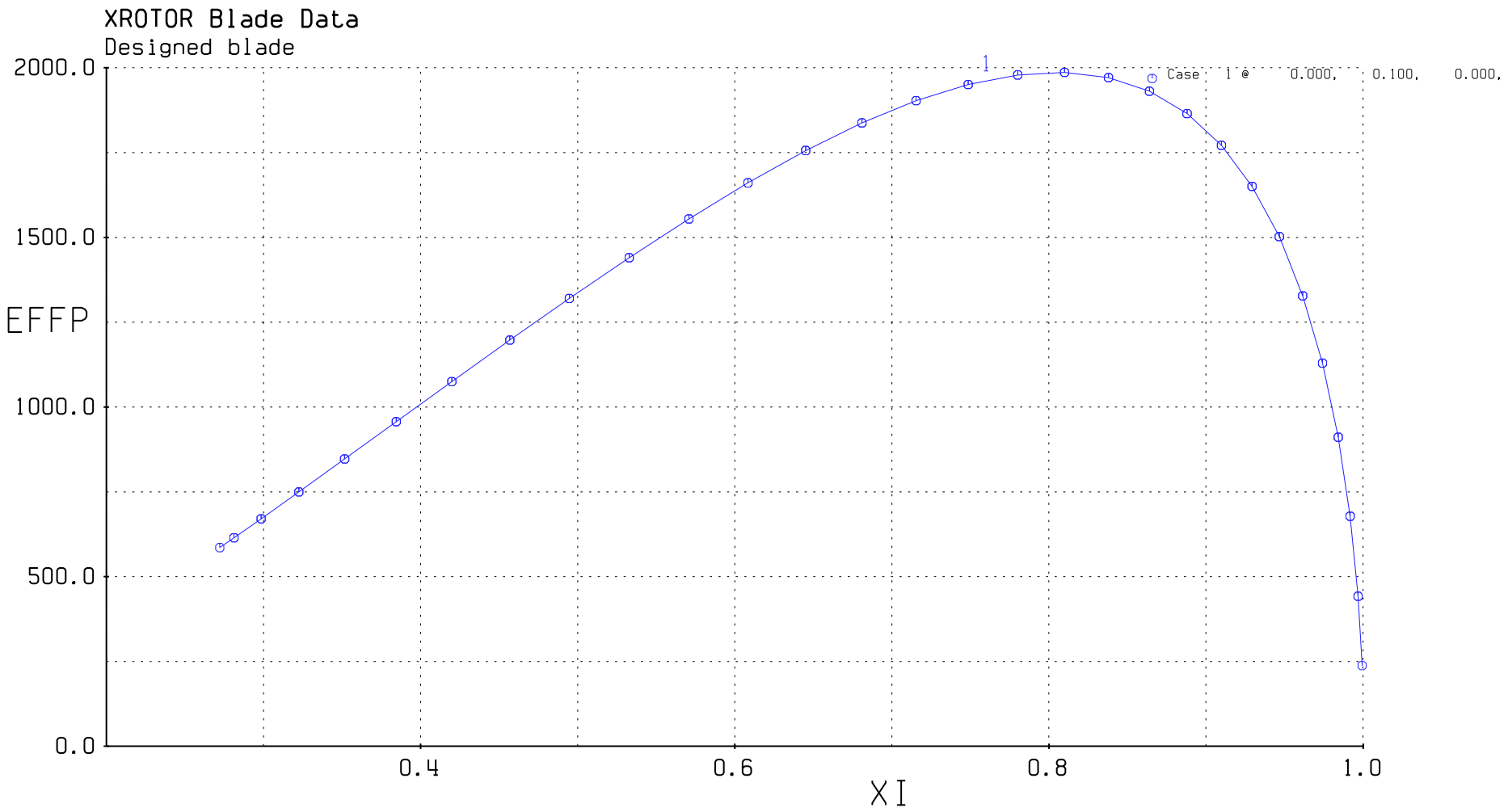

Designed blade
T kN= 2.1459
P kW= 57.6878
RPM = 4311.0
β tip = 0.000
Induced velocity at blade
Velocity induced
Circum Avg
V m/s
50.0
40.0
30.0
20.0
10.0
0.0
V axi
V tan
0.0
0.2
0.4
0.6
0.8
1.0
r/R


XROTOR Blade Data
Designed blade
Case 1 @ 0.000, 0.100, 0.000,
BE
XI
0.0
0.1
0.2
0.3
0.4
0.5
0.6
0.4
0.6
0.8
1.0
1


XROTOR Blade Data
Designed blade
Case 1 @ 0.000, 0.100, 0.000,
1
1.0
0.9
0.8
0.7
CL
0.4
0.6
0.8
1.0
XI


XROTOR Blade Data
Designed blade
Case 1 @ 0.000, 0.100, 0.000,
1.00
0.75
CD
0.50
0.25
0.00
-0.25
-0.50
0.4
0.6
0.8
1.0
XI
1


XROTOR Blade Data
Designed blade
2000.0
1500.0
1000.0
500.0
0.0
Ti+v
0.4
0.6
0.8
1.0
XI
1
Case 1 @ 0.000, 0.100, 0.000,


XROTOR Blade Data
Designed blade
2000.0
1500.0
1000.0
500.0
0.0
EFFP
0.4
0.6
0.8
1.0
XI
1
Case 1 @ 0.000, 0.100, 0.000,