

# Evolution of A Triad Twisted String Actuator for Controlling a Two Degrees of Freedom Joint to Improve Performance and Allow for Active Transmission Adjustment

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**Abstract**—actuated universal joint are used in a wide range of robotic applications, including mobile snake robots, snake-arm robots and robotic tails. Depending on the application and design constraints, these can use remote cable or fluid driven systems, or inline motors. In order to realise the benefits of inline actuation while keeping the system compact with a high power to weight ratio, an actuated universal joint was developed using an “antagonistic triad” of three twisted string actuators. However, the design of this system had numerous drawbacks in its current form, namely a limited angle range, poor accuracy due to the angular feedback sensors used, and issues with string failure due to mechanical design choices. In this publication, these drawbacks are addressed through design changes, and angle range was increased from  $\pm 14.5^\circ$  to  $\pm 0.0^\circ$  for a single axis, and  $\pm 6^\circ$  to  $\pm 0.0^\circ$  for a dual axis. These changes also allowed the introduction of active transmission adjustment, which allows for changing the transmission reduction during operation, which adjusts the maximum torque and velocity to adapt to different joint loads.

**Index Terms**—Flexible Robots, Force Control, Tendon/Wire Mechanism, Twisted String Actuator.

## I. INTRODUCTION

AUJ mechanisms are found in a wide range of robotic applications that require soft-rigid reconfigurable mechanisms, such as confined space inspection using continuum robots [1], highly manoeuvrable mobile snake robots [2], and biomimetic robot tails for stability [3]. These can either use inline actuators which directly move the joint [4]–[6], or cable/fluid driven systems that rely on a static “base” to house the actuators or compressors [7]. In the former case, this results in high torque requirements for the actuators, as they have to lift the mass of all the actuators in subsequent sections. In the latter case, space is required for the base, which is not practical in all applications, such as mobile robots.

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First developed by Würtz *et al.* [8] in 2010, the TSA uses two or more strings between two fixtures as a linear actuator. When one fixture is rotated (typically by an electric motor), the looped string twists into a helix, decreasing the distance between them. TSA actuators have been used for a hand orthosis [9], elbow joint [10] and foldable robot arm [11] among other functions.

Whereas alternative electric actuation systems, such as leadscrews, require the addition of gearing for significant increases in reduction (lower velocity, higher torque), which increases actuator size and mass, a TSA can increase reduction by reducing string thickness or string count, which slightly reduces actuator mass [12]. In the case of leadscrews, the reduction can be increased marginally by decreasing thread lead [13], but this can quickly come up against manufacturing tolerances and material stiffness and toughness limitations. The screw radius can also be increased, but this has the effect of increasing actuator size and mass in a similar fashion.

One of the main challenges of TSA is the reduction changes depending on the motor angle (and therefore actuator position). This is an inverse nonlinear relationship, where the function has a decreasing derivative as the motor angle increases [8], [12]. String compliance may also need to be considered under high force conditions, but this can be mitigated with accurate modelling [14], or a suitably robust control strategy that can ignore compliance in its system model [8], [12]. Since strings are not rigid, TSA can only impart force in tension, and 1 degree of freedom (DOF) single TSA actuator joints must make use of a spring return mechanism [15]–[17], which can limit actuator range since the spring force increases as maximum TSA force decreases [18]. Usman *et al.* [18] developed a linear force return mechanism ( $k = 0$ ) to partially mitigate this issue, but an ideal situation is to have a matching antagonistic force profile. This can be done by using a second TSA actuator which is synchronised with the first [10], [19]–[21]. Therefore, by adding a third TSA, a 2 DOF actuator can be realised without the use of springs.

The use of TSA as an actuator for an AUJ is an understudied area of research. Konda, Bombara, Chow, *et al.* [22] have proposed a similar design using a flexible core

with continuous curvature as opposed to a rigid universal joint, which outlines an open loop control solution for multi axis control using only two TSA experimentally demonstrated with a limited azimuthal axis range in polar coordinates. For the first time the authors demonstrate a robust closed loop control of an AUJ in both axes of motion, including the full azimuthal range of  $[0, 360]^\circ$ , using three TSA in an “antagonistic triad” configuration. The result is a light, compact AUJ design that has the potential to significantly improve upon exiting inline actuation options.

In this publication, the authors aim to improve on the prototype constructed in [23], previously published as a conference proceeding at the IEEE International Conference on Robotics and Automation (ICRA) 2022, by improving the joint angle range, string lifetime, and by introducing ATA, for adjustments to the dynamics of the system in real time both before and during operation. This publication will also conduct additional experiments with increased follower mass and joint velocity, include data to characterise each individual TSA under force control, and expand on the discussions in the conference proceedings with a focus on multi-segment operation.

#### A. A Twisted string actuator

First developed by Würtz, May, Holz, *et al.* [8] in 2010, TSA uses two or more strings between two fixtures as a 1 DOF linear actuator. When one fixture is rotated (typically by an electric motor), the strings twist into a helix, decreasing the distance between the fixtures, as shown in figure 1. Given the unwound length  $l_u$ , and the cross-section radius of the string  $r_s$  (or  $r_s + r_c$  when there are more than two strings, where  $r_c$  is the radius of a tangentially constrained circle drawn between the strings) as shown in figure 2, the actuator length is given by

$$l_s(\theta_s) = \sqrt{l_u^2 - \theta_s^2 r_s^2} \quad (1)$$

where  $\theta_s$  is the motor angle, as shown in figure 3.

This equation assumes an infinite string stiffness, so is only reasonably accurate under low tension. Although theoretically the stroke of the TSA can be the entire domain of  $[0, l_u]$ , in reality the thickness of the string prevents a geometric helix from forming once the helix pitch  $q < 4r_s$  (or  $q < 2nr_s$  for  $n$  strings). This limits the lower bound of the stroke as follows,

$$l_{min} = \frac{l_u}{\sqrt{\frac{\pi^2}{2} + 1}} \approx 0.46 l_u \quad (2)$$

or approximately 46% of  $l_u$  for a two string TSA.

#### B. Antagonistic Triad

As mentioned in the introduction, three TSA can operate an “antagonistic triad”, analogous to the antagonistic pairs for 1 DOF joints used in [10], [19]–[21]. These are arranged at three equidistant points around a circle of radius  $r$ , to form two equilateral triangles at either end

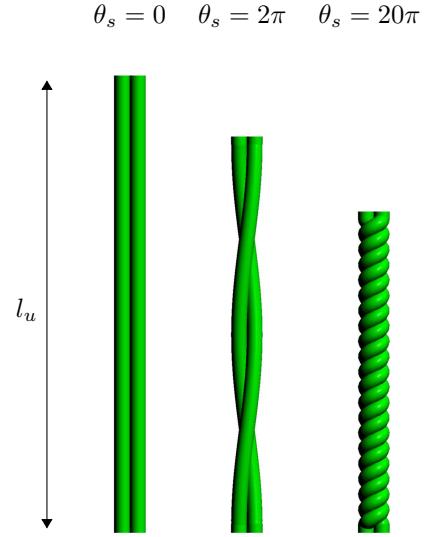


Fig. 1: The value of  $\theta_s$  increases the number of twists in a string bundle with a string length  $l_u$ .

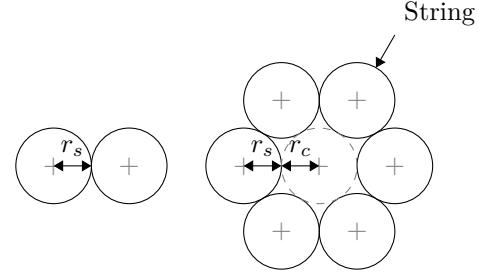


Fig. 2: The location of  $r_s$  and optionally  $r_c$  in a string bundle.

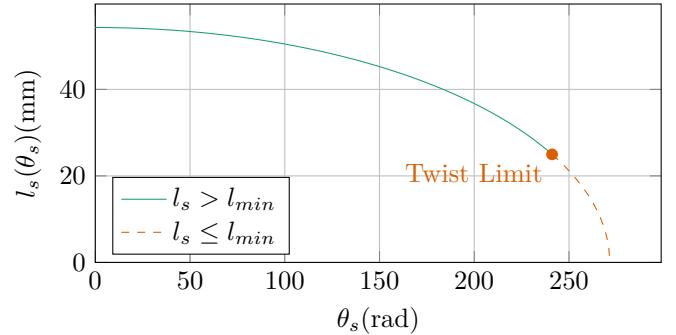


Fig. 3: TSA string length against motor angle with coefficients from table I.

of the linear actuators. The centers of the circles are connected along their normal axes to a universal joint, with a magnitude of  $l_1$  from the universal joint to the start of the actuators, and  $l_2$  from the universal joint to the end. This geometric construction is shown in figure 4. If  $\boldsymbol{\theta} = [\theta_1 \ \theta_2]^\top$  is defined as the rotation of the universal joint on the  $x$  and  $y$  axes in the coordinate system at the start of the actuators, such that the normal vectors are collinear when  $\boldsymbol{\theta} = [0 \ 0]^\top$ , then the magnitude for the

vector between each pair of points at either end of the actuator can be calculated using

$$\begin{aligned}\lambda_1(\boldsymbol{\theta}) &= \sqrt{a + 2l_1r \sin(\theta_2) \cos(\theta_1) + l_2^2} \\ \lambda_2(\boldsymbol{\theta}) &= \sqrt{a + b + c - d} \\ \lambda_3(\boldsymbol{\theta}) &= \sqrt{a - b - c + d}\end{aligned}$$

where:

$$\begin{aligned}a &= l_1^2 + 2l_1l_2 \cos(\theta_1) \cos(\theta_2) \\ b &= \sqrt{3}l_1r \sin(\theta_1) - l_1r \sin(\theta_2) \cos(\theta_1) + l_2^2 \\ c &= \sqrt{3}l_2r \sin(\theta_1) \cos(\theta_2) - l_2r \sin(\theta_2) \\ d &= \frac{\sqrt{3}r^2 \sin(\theta_1) \sin(\theta_2)}{2} - \frac{3r^2 \cos(\theta_1)}{2} - \frac{r^2 \cos(\theta_2)}{2} + 2r^2.\end{aligned}\quad (3)$$

where  $\Lambda(\boldsymbol{\theta}) = [\lambda_1(\boldsymbol{\theta}) \quad \lambda_2(\boldsymbol{\theta}) \quad \lambda_3(\boldsymbol{\theta})]$  is a vector function which outputs the magnitudes of each point pair, and therefore the lengths of each actuator, assuming both ends of each actuator can rotate freely on both  $x$  and  $y$  axes. The output of this equation is plotted in figure 5 for a domain of  $[-\frac{\pi}{2}, \frac{\pi}{2}]$ , with coefficients from table I.

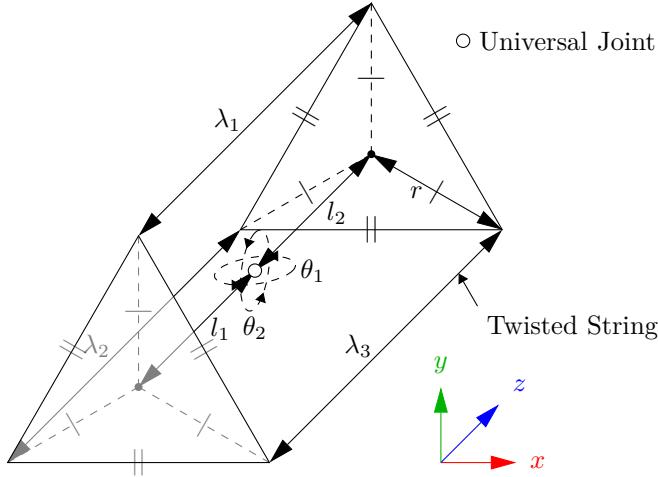


Fig. 4: Kinematic diagram of the antagonistic triad, where the universal joint rotation is defined by  $\theta_{1,2}$  on the  $y$  and  $x$  axes respectively, and the actuator lengths are defined by  $\lambda_{1,2,3}$  for the *top*, *left* and *right* strings.  $r$  and  $l_{1,2}$  define the anchor points of the strings.

## II. CONTROL SYSTEM

While position control for 1 DOF TSA joints, both spring return [24] and antagonistic [25], is possible, it proved too unstable to be used for the antagonistic triad. Therefore a force control system was designed for each TSA using a simple P controller, which a position control system would cascade down to, by converting the setpoint angle of the AUJ into three force setpoints for each TSA. This was done by first converting the angular velocity from the AUJ position PID controller to a setpoint AUJ torque using inverse dynamics [26]. A force optimisation algorithm from [27] was then implemented that takes the

desired AUJ joint torque as an input and outputs a tensile force setpoint for each TSA.

This control system functions because each TSA has a tensile force when  $l_s(\theta_s) < \lambda_n(\boldsymbol{\theta})$ , which is calculated as the difference in triad length and TSA contraction length multiplied by the load stiffness  $(\lambda_n(\boldsymbol{\theta}) - l_s(\theta_s))K_L$ . When all of the TSA tensile forces are equal, there is no torque on the AUJ, but when they are unequal, a torque is generated, and if these torques do not match the torques required by the current dynamics to maintain the AUJ orientation, the AUJ will rotate. As the tensile force is measured using load cells, the control system does not need to know the value of  $K_L$  since a setpoint force can be maintained using a P controller.

This control system can be split up into four functions:

- 1)  $C_1$  AUJ Position PID Controller with Acceleration Feedforward
- 2)  $C_2$  Inverse Dynamics
- 3)  $C_3$  TSA Force Optimisation Algorithm
- 4)  $C_4$  TSA Force P Controller

These functions are nested into a cascade function shown in figure 7.

### A. AUJ Position PID Controller with Acceleration Feed-forward

Firstly, a PID controller is used to generate a control signal  $\mathbf{u}$  with the desired AUJ angular position and velocity  $\mathbf{q}, \dot{\mathbf{q}}$  as the setpoint. The current AUJ angular position  $\boldsymbol{\theta}$  and velocity  $\dot{\boldsymbol{\theta}}$  as feedback, plus the addition of a feedforward term for the desired input acceleration  $\ddot{\mathbf{q}}$ .  $\dot{\mathbf{q}}$  and  $\ddot{\mathbf{q}}$  can be pre-programmed as part of a second order trajectory profile, or can be calculated from a ramp generator when  $\mathbf{q}$  is manually controlled.

In the discrete implementation used for fixed step simulation and experimental system control, the integral term of the PID is replaced by the trapezoidal rule.

### B. Inverse Dynamics

The control signal  $\mathbf{u}$  from the PID controller is then converted to the desired AUJ torque  $\boldsymbol{\tau}$ . This is achieved using the Euler-Lagrange formulation which takes into account the dynamic properties of the AUJ in its current state, as shown in figure 6. For simplicity, the mass of the universal joint is ignored in the calculations and the value of the follower COM relative to the universal joint pivot  $\rho$  only has a  $z$  component,  $\rho_3$ , which is included in the experiments in section V-C4 but set to zero for all other experiments. The torque is calculated as  $\boldsymbol{\tau} = D(\boldsymbol{\theta})\mathbf{u} + C(\boldsymbol{\theta}, \dot{\boldsymbol{\theta}})\dot{\boldsymbol{\theta}} + G(\boldsymbol{\theta})$  where the functions  $D$ ,  $C$ , and  $G$  are given in (4).

### C. TSA Force Optimisation Algorithm

This uses the *inverse force transformation* algorithm from [27] with the jacobian in equation 3 to select an optimal force vector from the desired joint torque. Here it is presented in an unexpanded and more general form,

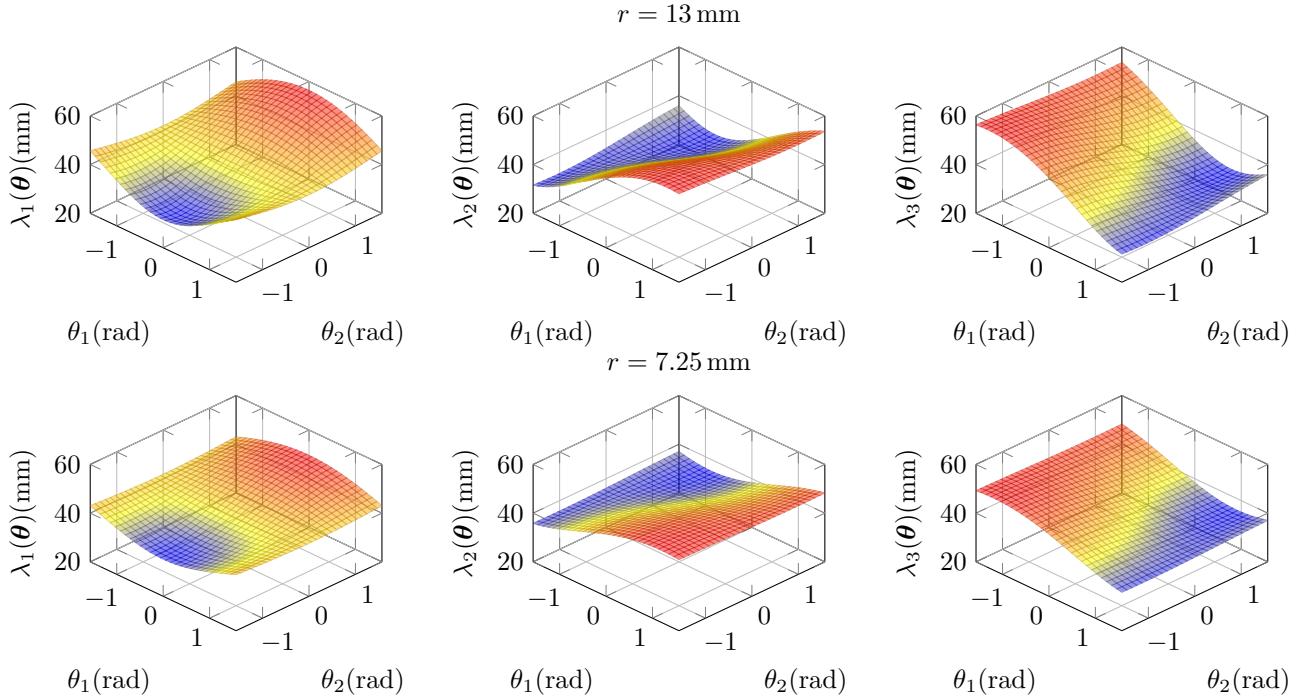


Fig. 5: Surface plots of each element of the vector function  $\Lambda(\theta)$ , assuming coefficient values from table I. For every pair of universal joint angles  $\theta$ , there is a corresponding vector of actuator lengths  $\Lambda(\theta)$ . Note that  $\lambda_2$  and  $\lambda_3$  are symmetric.

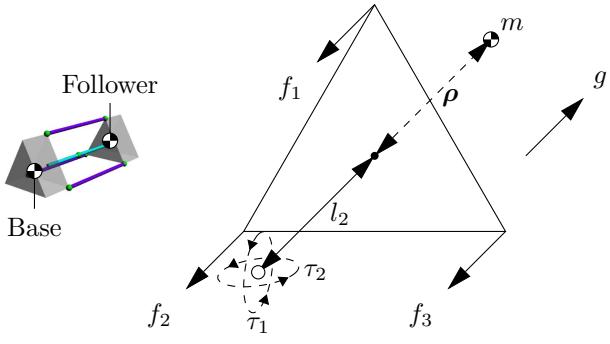


Fig. 6: Dynamics of an antagonistic triad based on figure 4, where the follower mass is  $m$ , the follower center of mass (COM) relative to the universal joint pivot is  $\rho$ , the forces on each anchor point are  $f_{1,2,3}$  and the universal joint torque is  $\tau_{1,2}$ .

$$\begin{aligned} J_\Lambda &= \begin{bmatrix} \frac{\partial \lambda_1}{\partial \theta_1} & \frac{\partial \lambda_2}{\partial \theta_1} & \frac{\partial \lambda_3}{\partial \theta_1} \\ \frac{\partial \lambda_1}{\partial \theta_2} & \frac{\partial \lambda_2}{\partial \theta_2} & \frac{\partial \lambda_3}{\partial \theta_2} \end{bmatrix} \\ \gamma(i) &= -J_{\Lambda_{-i,*}}^{-T} \left( J_{\Lambda_{i,*}}^T f_{min} + \tau \right) \\ F(\tau, \theta) &= \begin{bmatrix} f_{min} & \gamma(2)_1 & \gamma(3)_1 \\ \gamma(1)_1 & f_{min} & \gamma(3)_2 \\ \gamma(1)_2 & \gamma(2)_2 & f_{min} \end{bmatrix}. \end{aligned} \quad (5)$$

A force matrix  $F$  is created from the torque input  $\tau$ , jacobian  $J_\Lambda$  from the vector function  $\Lambda$  as defined in

equation 3, and minimum force constant  $f_{min} \cdot f_{ii}$  is equal to  $f_{min}$ , while the other elements in the column are based on a calculation using  $J_{\Lambda_{-i,*}}$  where  $-i$  is a row removed from the matrix. Algorithm 1 is then used to create output force vector  $f$ , which minimises the net force on all TSA while producing the desired output torque on the universal joint.

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**Algorithm 1** Selects one column of  $F$  to be the output force vector  $f$ , where  $\top$  and  $\perp$  are boolean *true* and *false* respectively, and  $f_{*,i}$  is the  $i$ th column of  $F$ .

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1:  $s \leftarrow [\top \ \top \ \top]$ 
2: if  $f_{23} > f_{min}$  then  $s_2 \leftarrow \perp$  else  $s_3 \leftarrow \perp$  end if
3: if  $f_{31} > f_{min}$  then  $s_3 \leftarrow \perp$  else  $s_1 \leftarrow \perp$  end if
4: if  $f_{12} \geq f_{min}$  then  $s_1 \leftarrow \perp$  else  $s_2 \leftarrow \perp$  end if
5: for  $i = 1$  to  $3$  do
6:   if  $s_i \rightarrow \top$  then  $f_{set} \leftarrow f_{*,i}$  end if
7: end for
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#### D. TSA Force Proportional Controller

The selected forces are then used as an input to a P controller with gain  $k_{ps}$  using the measured load cell forces  $f_{act}$  as feedback. The output from this can then be used to control the top, left and right TSA motors, corresponding to the actuators in figure 4. A deadband controller was used to compensate for motor limitations, which is further discussed in section V-B1.

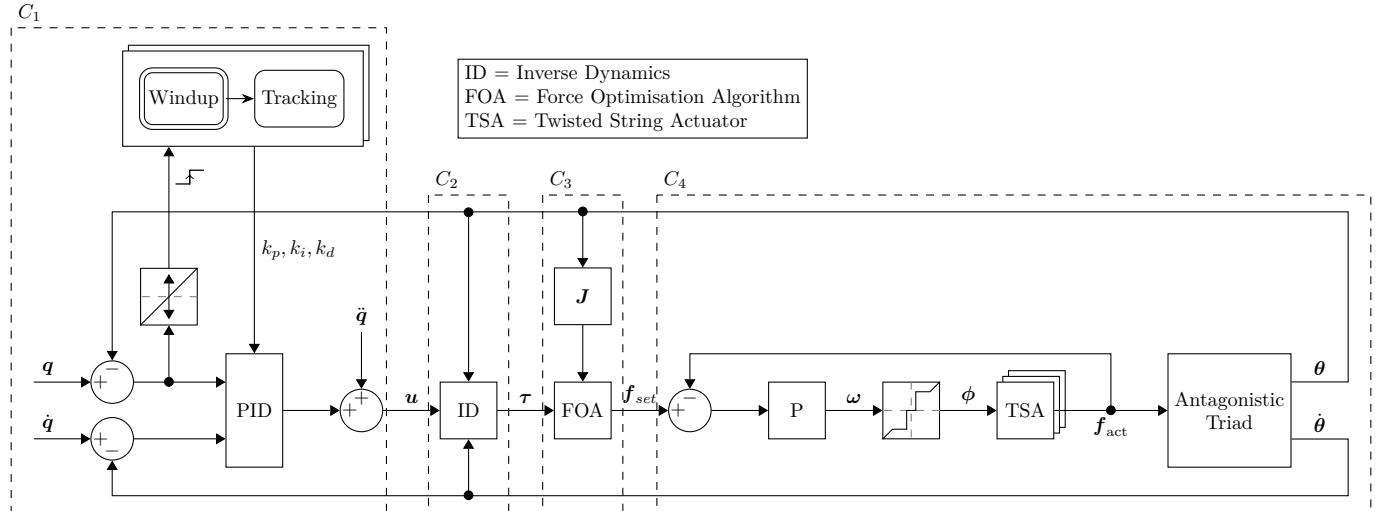


Fig. 7: Block diagram of the complete experimental control system, excluding the hardware velocity controllers for the motors. Dashed boxes correspond to the control functions  $C_1 \dots 4$  specified in section II.

$$\begin{aligned}
 D(\boldsymbol{\theta}) &= \text{diag} \left( \left[ I_{22} \cos^2 \theta_2 - I_{33} \cos^2 \theta_2 + I_{33} + l_2^2 m \cos^2 \theta_2 + 2l_2 m \rho_3 \cos^2 \theta_2 + m \rho_3^2 \cos^2 \theta_2 \right]^\top \right) \\
 C(\boldsymbol{\theta}, \dot{\boldsymbol{\theta}}) &= \begin{bmatrix} \frac{(-I_{22} + I_{33} - l_2^2 m - 2l_2 m \rho_3 - m \rho_3^2) \sin(2\theta_2) \dot{\theta}_2^2}{2} & \frac{(-I_{22} + I_{33} - l_2^2 m - 2l_2 m \rho_3 - m \rho_3^2) \sin(2\theta_2) \dot{\theta}_1^2}{2} \\ \frac{(I_{22} - I_{33} + l_2^2 m + 2l_2 m \rho_3 + m \rho_3^2) \sin(2\theta_2) \dot{\theta}_1^2}{2} & 0 \end{bmatrix} \\
 G(\boldsymbol{\theta}) &= [-gm(l_2 + \rho_3) \sin \theta_1 \cos \theta_2 \quad -gm(l_2 + \rho_3) \sin \theta_2 \cos \theta_1]
 \end{aligned} \tag{4}$$

### III. SIMULATION DESIGN

To design and refine the parameters of the control system, a Simscape Multibody™ model of the antagonistic triad and control system was created in MATLAB®/Simulink™. This allowed for model design coefficients  $l_{1,2}$  and controller gains  $k_p, k_i, k_d, k_{p_s}$  to be optimised in order to have the most stable control within design limits, as shown in table II.

In the spirit of [8], each TSA was modelled as a state-space system with saturation function  $\text{sat}_x^y z = \max(x, \min(y, z))$ , and a state-space representation given by

$$\begin{aligned}
 h(\theta_s) &= \frac{\theta r_s^2}{\sqrt{l_u^2 - \theta_s^2 r_s^2}} \\
 k(\theta_s, \boldsymbol{\theta}) &= \lambda_n(\boldsymbol{\theta}) - \sqrt{l_u^2 - \theta_s^2 r_s^2} \\
 \dot{\boldsymbol{x}} &= \begin{bmatrix} x_2 \\ -\frac{K_L}{J} h(x_1) k(x_1, \boldsymbol{\theta}) - \frac{C}{J} \text{sgn}(x_2) \end{bmatrix} + \begin{bmatrix} 0 \\ \frac{K_t}{J} \end{bmatrix} u \\
 y &= K_L \text{sat}_0^\infty k(x_1, \boldsymbol{\theta}),
 \end{aligned} \tag{6}$$

which takes motor current  $u$  as an input and outputs  $y$  as the TSA tension force, where  $J$  is the motor inertia,  $C$  is the motor coulomb friction (modified from viscous friction as the motor only has dry friction),  $K_t$  is the motor torque constant, and  $K_L$  is the load stiffness. As the original definition is for a fixed load  $l_u$  distance from

the motor a modified model is required which takes into account the varying length between the motor and load defined by  $\Lambda(\boldsymbol{\theta})$ .

The saturation function is used to prevent incorrect compression forces when the string is slack. All of the motor coefficients were taken from the Faulhaber 1724TSR datasheet [28] as this is the motor used in the experimental model. An estimated value is used for the load stiffness, this was chosen to be a high number as the model is expected to be very stiff.

The state space model was then adapted to include constraints on motor velocity and acceleration set by the motor controller in order to keep the motor within design limits  $\dot{\boldsymbol{x}}' = [\text{sat}_{-\omega_s}^{\omega_s} \dot{x}_1 \quad \text{sat}_{-\alpha_s}^{\alpha_s} \dot{x}_2]^\top$ , which replaces  $\dot{\boldsymbol{x}}$  with  $\dot{\boldsymbol{x}}'$ , which contain saturation functions for maximum motor velocity  $v_s$  and acceleration  $\alpha_s$ .

Based on a combination of simulation modelling, motor parameters, and physical design constraints, the model parameters for the first version were set to the values in table I. Following experimental results and issues encountered from the first version, the model parameters were altered to improve performance in the second version, which can also be found in the table.  $l_2$  was set to zero, as this produced the most stable control in simulation.

TABLE II: PID gains in the simulation and experiment.

Gain	Value	
	Simulation	Experiment*
$k_p$	800	$3 \times 10^4$
$k_i$	3000	350
$k_d$	50	50
$k_{ps}$	19	100

\* Tracking mode, see section V-B2.

#### IV. EXPERIMENTAL DESIGNS

TABLE I: Model coefficients.

Coefficient	Value	Coefficient	Value
$l_1$	41.8 mm	$J$	$1 \times 10^{-6} \text{ kg m}^{-2}$
$l_2$	0 mm	$K_L$	$1000 \text{ N m}^{-1}$
$r$	13 mm/6 mm*	$f_{min}$	3 N
$l_u$	41.8 mm	$\omega_s$	$441.9 \text{ rad s}^{-1}$
$r_s$	200 $\mu\text{m}$	$I_s$	0.19 A
$m$	72.619 13 g	$K_t$	0.0263 N m A $^{-1}$
$C$	0.1315 N mm	$\tau_s$	4.5 mN m
$\alpha_s$	$1 \times 10^5 \text{ rad s}^{-2}$	$\rho$	[0 0 0] mm

Coefficient	Value
$I$	$\begin{bmatrix} 3 \times 10^{-5} & 0 & 0 \\ 0 & 3.2 \times 10^{-5} & 0 \\ 0 & 0 & 1.4 \times 10^{-5} \end{bmatrix} \text{ kg m}^{-2}$

\* Version 1/Version 2.

For the experimental validation, a single segment physical prototype of the mechanism was constructed with coefficients from table I as design parameters.

##### A. Version 1

The first version of the prototype consisted of three Faulhaber 1724TSR brushed DC motors attached to the base segment with a 3D printed cowling. These were directly connected to string clamps made out of machined brass which the string, made of SeaKnight BLADE 0.2 mm monofilament nylon fishing line, was secured to with two M1.6 grub screws at each end. The strings then looped through a capstan bolt which was connected to a FUTEK LCM100  $\pm 9.8 \text{ N}(1\text{kg})$  micro miniature load cell, selected for their small size with a diameter of only 9.5 mm, which was then connected to the follower segment via a universal joint, to allow the load cell sensing axis to remain collinear with the string during operation so the correct force was measured. Each of these created a TSA mechanism for the AUJ. To complete the AUJ, the base and follower segments were also connected by two 3D printed shafts connected by a universal joint, with its  $x$  and  $y$  axes on the same plane as the  $x$  and  $y$  axes of the universal joints, making  $r_2 = 0$ . To sense the orientation of the follower segment, a Bosch BN0080 inertial measurement unit (IMU) would read the orientation of the follower segment, and therefore the orientation of the AUJ. It was intended for the IMU to measure orientation of the end effector to be measured in three dimensions, so the AUJ orientation could be calculated by taking the difference between the base segment

orientation and the follower segment orientation, however issues the magnetometer on the BNO080 meant only two dimensions could be measured, and the AUJ needed to be in a vertical orientation for this to be possible.

The prototype was controlled by a National Instruments MyRIO-1900, with the main control loop operating at a frequency of 5 Hz. The motors were controlled by Fauhaber MCDC3002 motor controllers, which could interface with the MyRIO via the USB port, using a USB to serial converter. The load cell signals, with a sensitivity of  $1 \text{ mV V}^{-1}$  [29], were first amplified using Flyde FE-359-TA instrumentation amplifiers, and then read using an external AD7606 ADC with a 14-bit resolution [30], with the data being transmitted to the MyRIO via an SPI bus.

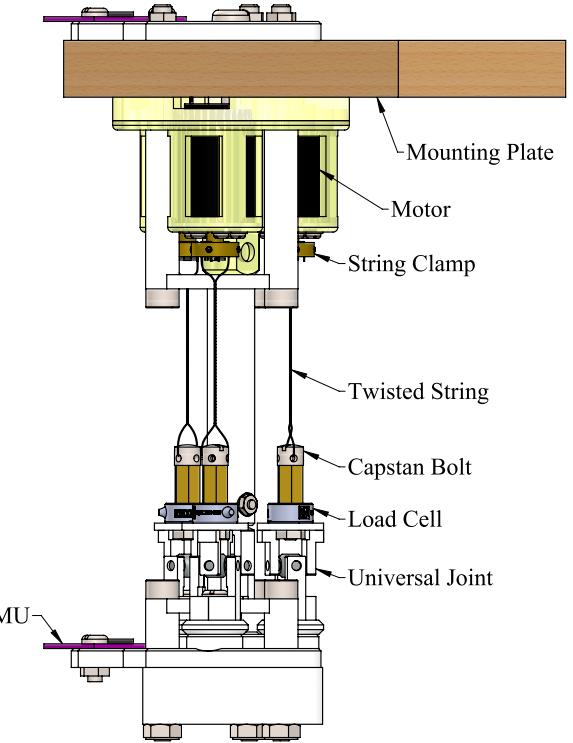


Fig. 8: Schematic of the version 1 single segment physical prototype antagonistic triad with labelled components.

##### 1) Discussion:

a) *Monofilament Torsion Fatigue & String Breakage:* TSA string breakage was common during the experiments in version 1, this was often due to a loss of control stability during tuning, or operation beyond previously explored motor angle or load limits. However, failure rates were observed to increase with the number of working cycles (winding and unwinding) of each TSA string, particularly when taken to high motor angles or loads. Nylon monofilament, as was used for the TSA string, is susceptible to torsion fatigue [31], [32] which reduces tensile strength [33]. This decrease in strength could be a potential explanation for the increase in failure rate. There was also the route and securing method of the string itself, with the four grub screws securing the string to the string clamp, and also

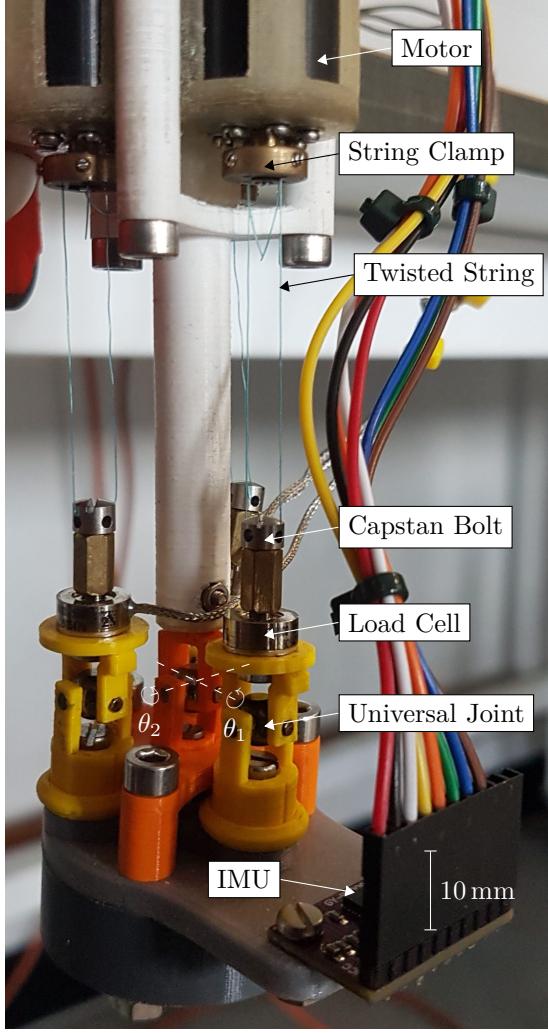


Fig. 9: Annotated photograph of the single segment physical prototype antagonistic triad, with the roll  $\theta_1$  and pitch  $\theta_2$  axes marked.

passed close to six potentially sharp edges, four on the corners of the string clamp and two on the capstan bolt, as shown in figure 10. These locations could weaken the string, by either an edge biting into it when under high tension, or when being clamped by the grub screws.

The cause of the string break could be determined by the location of the break. If it was located at or near one of the grub screws or biting edges, then it was likely to be caused by those. If it was located on the “active” part of the string that twists, then torsion fatigue is likely to be the culprit. To somewhat mitigate the biting and clamping issue, polytetrafluoroethylene (PTFE) sleeving was added around the string near those locations, and this did result in a significant reduction in string breakages due to edge biting or clamping.

The experiments in [12] used 0.2 mm Dyneema<sup>®</sup> and Fast Flight<sup>®</sup> Plus polyethylene (PE) polyfilament, and were able to achieve over 1000 working cycles before failure at a load of  $\leq 20$  N at a motor angle of nearly 300 rad. Use of nylon monofilament was due to difficulties assembling

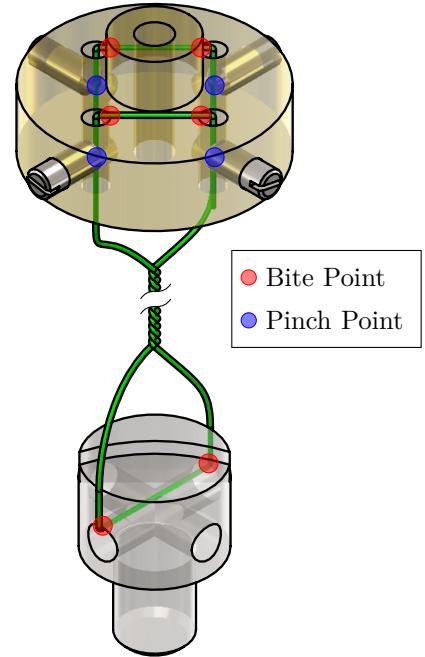


Fig. 10: Cutaway of a single TSA string assembly, including the capstan and string mount, highlighting the “pinch” points and “bite” points that could cause weaknesses in the string material, which may lead to premature failure.

the current design using polyfilament, as slicing the ends usually resulted in individual fibres splaying, which made it impossible thread through the string clamp.

#### B. Improving AUJ Angle Range & Enabling Active Transmission Adjustment by Reducing Triad Radius

As can be seen in figure 11, decreasing the triad radius  $r$  decreases the stroke range of each TSA for a given AUJ angle range. This increases the AUJ angle range limits without having to increase  $f_{min}$  as was done in section V-C2.

If  $r$  or the AUJ angle range is small enough, this allows for active transmission adjustment, where increasing or decreasing  $f_{min}$  shifts the active TSA stroke “window” to the right or left along the maximum force and maximum velocity curves from figure 28, as shown in figure 12. This would allow the performance of the AUJ to be adjusted during operation, which could be useful in some applications, such as a mobile snake robot which could increase AUJ angular velocity when all segments are on the ground, and sacrifice angular velocity in some segments for increased AUJ torque when they are off the ground and therefore subject to gravitational forces.

An issue with a small triad radius is the difficulty of accommodating the central shaft with the universal joint. This can be solved by using a wide universal joint with a hollow spider, which allows the three TSA to pass through

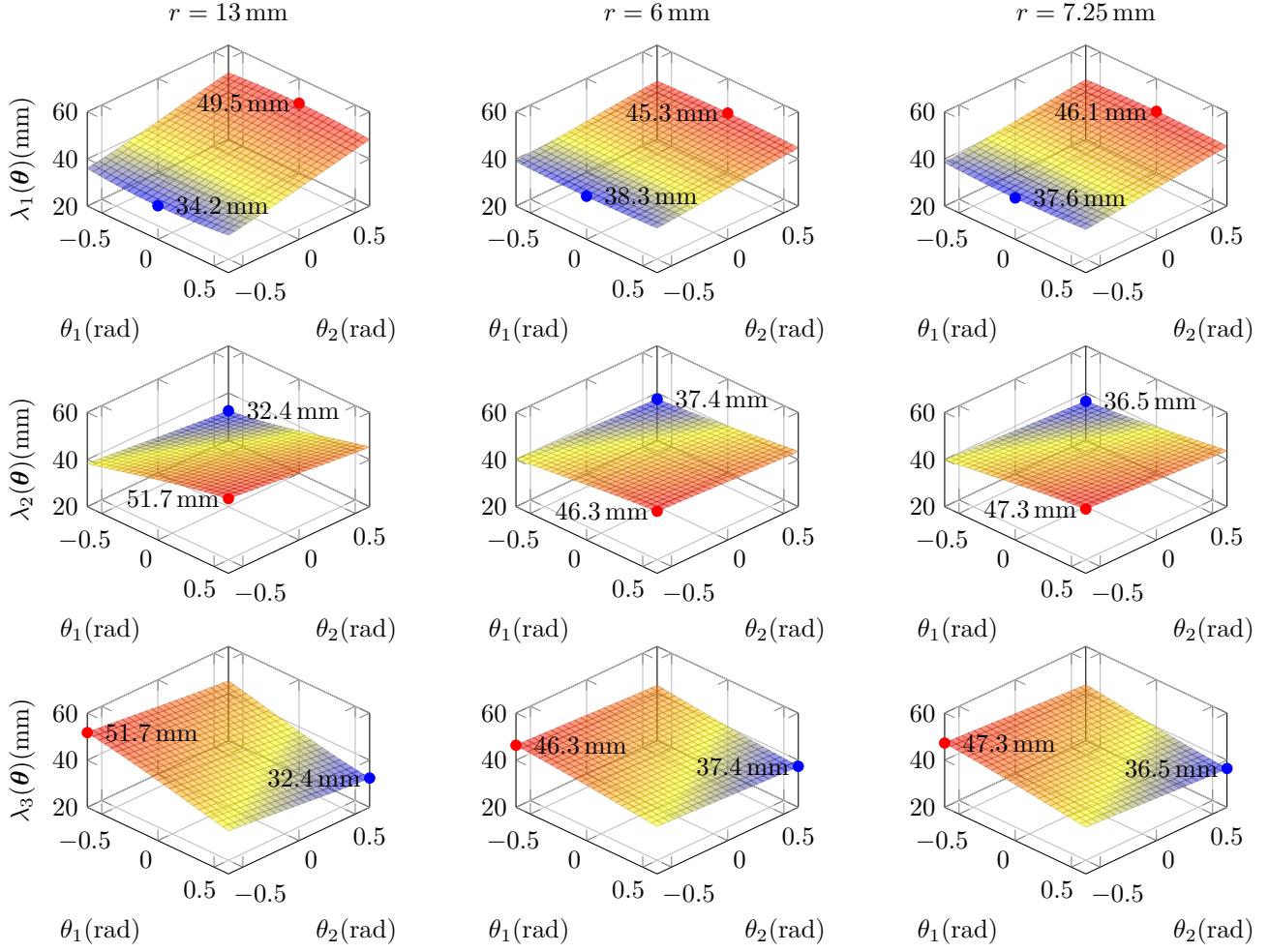


Fig. 11: Surface plots, minima and maxima for the lengths of  $\lambda_{1,2,3}$  at  $r = 13 \text{ mm}$ ,  $r = 10 \text{ mm}$  and  $r = 5 \text{ mm}$  in the range  $[-\frac{\pi}{5}, \frac{\pi}{5}]$ . As  $r$  decreases, the difference between the minima and maxima also decreases. This means that a lower value of  $r$  requires a smaller stroke range of the TSA for a given AUJ angle range.

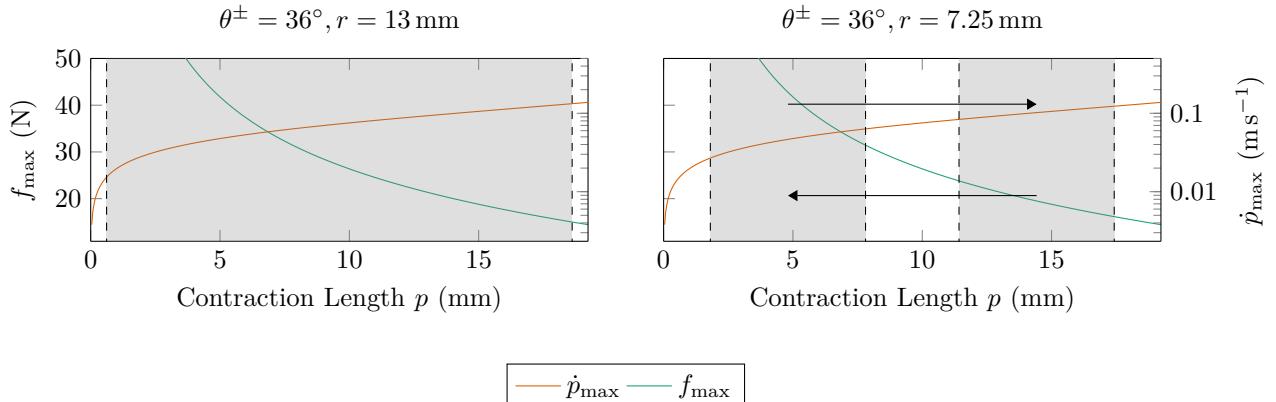


Fig. 12: By decreasing  $r$ , not only can the AUJ angle range be increased by reducing the TSA stroke range, which is marked in grey, but by adjusting  $f_{\min}$ , the transmission ratio of the TSA can be altered. Reducing  $f_{\min}$  increases the maximum TSA force  $f_{\max}$  while reducing the maximum TSA stroke velocity  $\dot{p}_{\max}$ . Conversely, increasing  $f_{\min}$  reduces  $f_{\max}$  and increases  $\dot{p}_{\max}$ . This can be used to actively modify the dynamic properties of the AUJ during operation.

the middle of the joint and the central shaft is not required, as shown in figure 13.

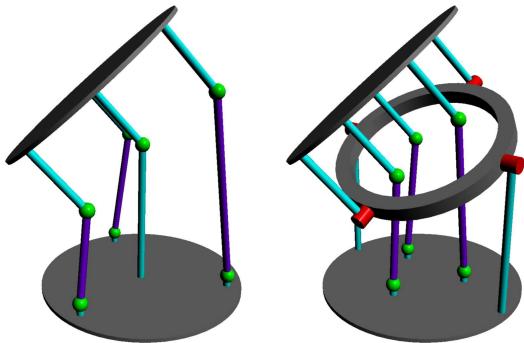


Fig. 13: An AUJ with a central universal joint, and one with a hollow spider. The hollow spider allows  $r$  to be decreased as space is no longer required for a central universal joint.

*a) BNO080 AUJ Orientation Sensor:* Using the BNO080 as the AUJ orientation sensor proved to be challenging. The magnetometer was unusable within the indoor experimental environment even after several calibration attempts due to hard and soft iron biases from other components and objects near the experimental area, so the universal joint angle would not be able to be calculated when the gravity vector is not orthogonal to the universal joint DOF, hence the experiments had the AUJ in a vertical orientation. The accelerometer data from the IMU, used to calculate the AUJ orientation, was of poor resolution as can be seen in figure 14.

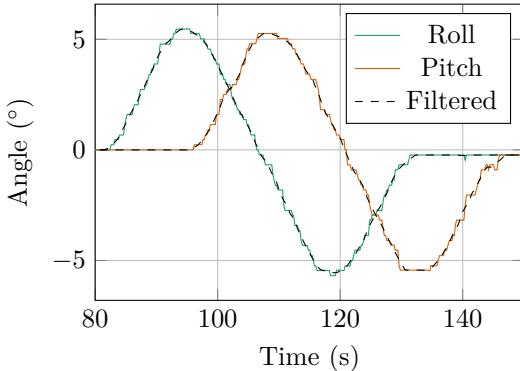


Fig. 14: Graph showing the AUJ orientation calculated from the raw IMU data, and smoothed data with the Savitsky-Golay filter applied.

### C. Version 2

Version 2 of the design was designed to address the issues raised in section IV-A1:

- Improving AUJ Angle Range & Enabling Active Transmission Adjustment by Reducing Triad Radius** -  $r$  was reduced from 13 mm to 6 mm using of a pair of spur gears to offset the axis of the motors, as shown in figure ???. This necessitated a

modification of the universal joint to a “hollow spider” design since there is no longer space for a suitably large central shaft, as shown in figure ??.

- Monofilament Torsion Fatigue & String Breakage** - this was mitigated with the following two improvements:

- The monofilament (Seaknight BLADE 0.2 mm) string was exchanged for a braided polyfilament (Dyneema® Micro Tex 0.2 mm). This is similar to the string used in [12]. Modifications to the string clamp to allow for easier assembly enabled polyfilament string to be used, as shown in figure ??.
- The string clamp and capstan design were modified to remove pinch and bite points to reduce the chance of premature string failure, as shown in figure ??.

- BNO080 AUJ Orientation Sensor** - The hollow spider universal joint design allowed for potentiometers (Bourns PDB08) to be attached to each axis, allowing the AUJ orientation to be determined directly. The analog voltage returned from the potentiometers means their resolution is determined by the analog to digital converter (ADC) on the MyRIO (12 Bits[34]), and they are unaffected by magnetic interference or gravity. As long as the potentiometers are sufficiently calibrated and filtered to reduce sliding noise, they will provide more precise and accurate AUJ orientation data.

Because  $r$  is now less than the diameter of the 1724TSR (8.5 mm[28]), it was not possible to drive the strings directly from the motors, so a pair of spur gears is required to drive an offset shaft at  $r = 6$ . The offset shaft was secured to base segment by two bearings and a locking nut, as shown in figure ???. It includes a new string clamp design with filleted holes for the ends of the string, which serve two purposes, to remove any sharp edges (“bite points” as defined in figure 10) and to guide frayed ends of the string into the hole, which allows for the use of polyfilament. The string can then be knotted at either end or tied together to secure it to the clamp. Since  $l_s(r_s)$  is an even function, the fact that the gear pairs reverse the direction of the motor is immaterial. Figure ?? shows a schematic and figure ?? shows a labelled photograph of the modified design.

## V. EXPERIMENTAL RESULTS

### A. Force Proportional Controller & Motor Characterisation

Before verifying robust control of the AUJ, the control of each individual TSA needed to be tested in order to characterise the performance of the motors and to ensure the inner loop control system was robust. This would involve selecting the control strategy that gave the best performance. A test force trajectory consisting of a smooth ramp followed by a sine wave was fed into a single tsa force setpoint as  $f_{set}(t)$ . The inner loop operated at a frequency of 100 Hz. Each control strategy was tested, and in the

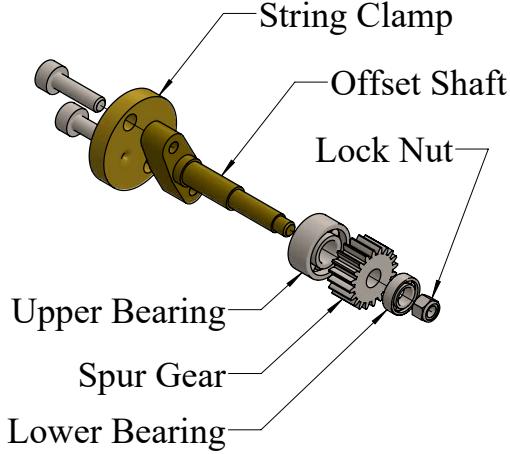


Fig. 15: Exploded diagram of the offset shaft with labelled components.

end the velocity control strategy proved most optimal, as is shown in figure 20. The results of a sinusoidal force trajectory for the velocity control strategy is shown in figure 21. As can be seen in the inset, there is a nonlinear friction effect which causes a sawtooth like effect on the measured force. This means the motor angle has to be constantly adjusted even with a constant force input, in order to maintain the setpoint force.

1) *Direct current controller*: This strategy takes a pre-set velocity  $\dot{\theta}_{\text{set}} \in [0, 4220] \text{ r min}^{-1}$  and uses a hardware PI controller with velocity feedback  $\dot{\theta}_{\text{act}}$  to generate a control voltage. This voltage  $V$  is then multiplied by the signum of the signal from the proportional force controller in order to ensure the motor spins in the right direction, and then passed to a current limiter with the current error (the proportional force controller signal with the actual current  $I_{\text{act}}$  subtracted) as the limit, before being sent to the motor. This ensures the motor stops spinning when the target current is reached.  $V_{\text{lim}}$  is calculated as

$$\begin{aligned}\epsilon_{\dot{\theta}} &= \dot{\theta}_{\text{set}} - \dot{\theta}_{\text{act}} \\ \epsilon_f &= f_{\text{set}} - f_{\text{act}} \\ V_{\text{lim}} &= \left( k_{p_c} \epsilon_{\dot{\theta}} + k_{i_c} \int_0^t \epsilon_{\dot{\theta}} \right) \text{sgn}(k_{p_s} \epsilon_f) \zeta((k_{p_s} \epsilon_f) - I_{\text{act}}),\end{aligned}\quad (7)$$

where  $k_{p_c}$  and  $k_{i_c}$  are the hardware controller PI constants,  $k_{p_s}$  is the force P constant,  $\dot{\theta}_{\text{set},\text{act}}$  and  $f_{\text{set},\text{act}}$  are the TSA motor velocity and string tension setpoints and responses respectively, and  $\zeta(\dots) \in [0, 1]$  is an unknown hardware function that limits the motor voltage depending on the current error. A block diagram of this strategy is shown in figure 17.

2) *Proportional Current Controller*: This strategy is a more direct method of current control, using a software P controller to directly set the voltage of the motor, using

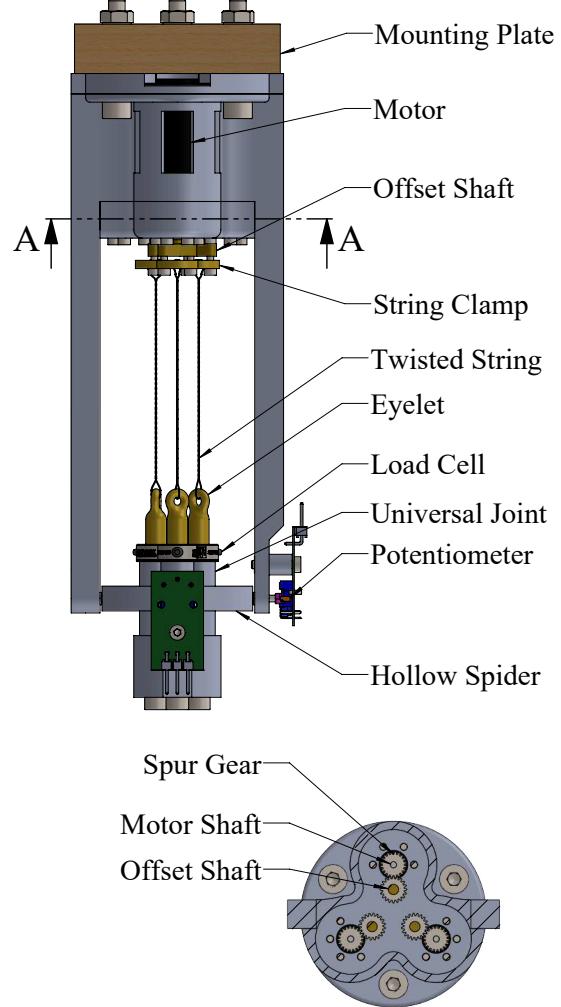


Fig. 16: Schematic of the version 2 single segment physical prototype antagonistic triad with labelled components.

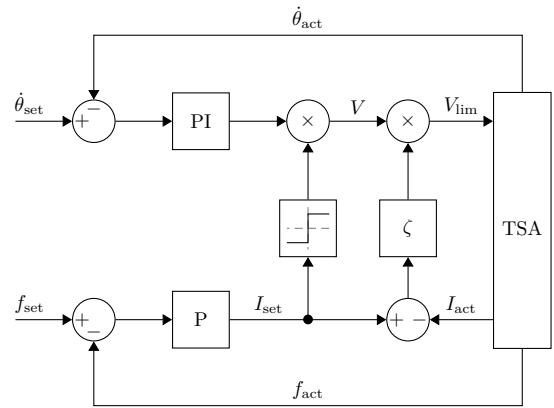


Fig. 17: Block diagram of the direct current controller.

the MCDC 3002 as simply a power amplifier. In this case

the current error is passed directly to a P controller which has its output limited to prevent damage to the motors.  $V$  is calculated as  $V = k_{p_s} \epsilon_f - I_{act}$ . A block diagram of this strategy is shown in figure 18.

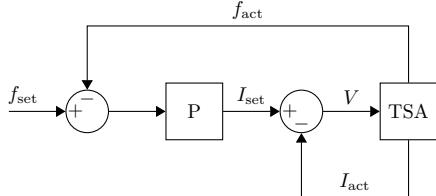


Fig. 18: Block diagram of the proportional current controller.

3) *Velocity Control*: This strategy simply uses the result of the P controller as a velocity setpoint using the hardware velocity PI controller.  $V$  is calculated as  $V = k_{p_c} (k_{p_s} \epsilon_f - \dot{\theta}_{act}) k_{i_c} \int_0^t (k_{p_s} \epsilon_f - \dot{\theta}_{act})$ . A block diagram of this strategy is shown in figure 19.

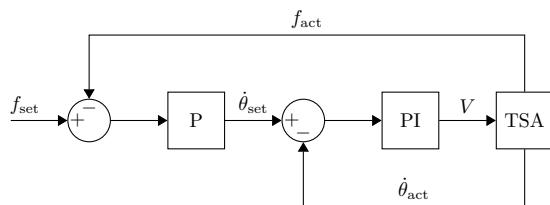


Fig. 19: Block diagram of the velocity controller.

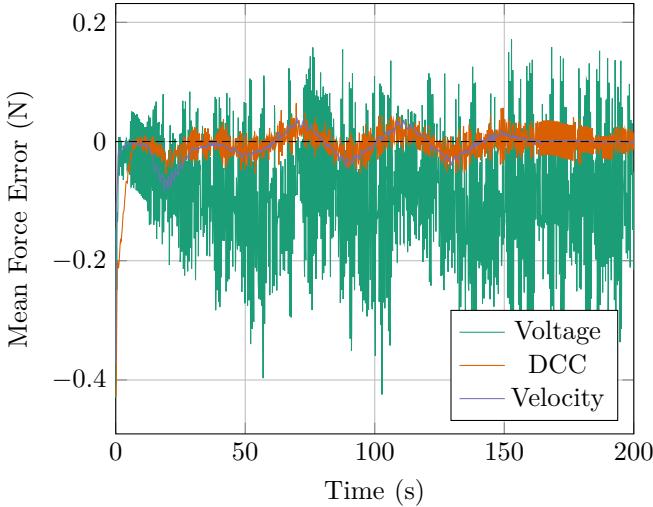


Fig. 20: Mean tracking error for each control strategy.

### B. Modifications to Control System

1) *Experimental Velocity Control with Deadband Compensation*: Due to a controller deadband within  $\pm 10 \text{ min}^{-1}$ , an adjustable deadband compensator is used,

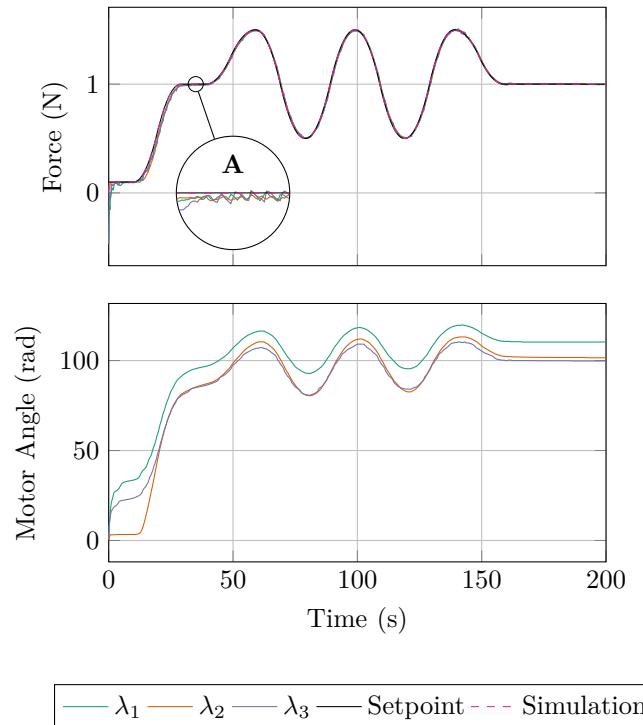


Fig. 21: Load cell, motor angle, and motor velocity signal plot from each string using the velocity control strategy for an experimental trajectory. Inset A shows a zoomed in portion of the graph, illustrating the nonlinear phenomenon noticed in the load cell measurements.

$$\phi_i(\omega_i) = \begin{cases} 10 & h < \omega_i \leq 10 \\ -10 & -10 \leq \omega_i < -h \\ 0 & -h \leq \omega_i \leq h \\ \omega_i & \text{otherwise} \end{cases} \quad (8)$$

where  $\phi_i$  is the compensator for the controller  $i$ . An adjustment value  $h \in [0, 10]$  changes the threshold at which the compensator switches on and off, allowing for a small deadband to remain.

2) *Windup & Tracking States*: When the mechanism is started with the TSA in a completely unwound state, before it can begin tracking a motion trajectory, the TSA strings must “wind up” to closely match the initial state of  $\mathbf{f}$ . During this phase, the outer PID gains  $k_p, k_i$  are unsuitable and can result in damage to the mechanism. To mitigate this, two sets of PID gains are chosen, one for the windup state ( $k_p = 800, k_i = 3000$ ), and another for the tracking state ( $k_p = 3 \times 10^4, k_i = 350$ ), which the windup state transitions to once suitable stability is achieved. This transition trigger is defined on a per-axis basis, as the first zero crossing of the angle error (as  $\mathbf{q} = 0$  this is effectively  $\boldsymbol{\theta}$ ). A graph showing the difference this state change makes to the AUJ orientation is shown in figure 22.

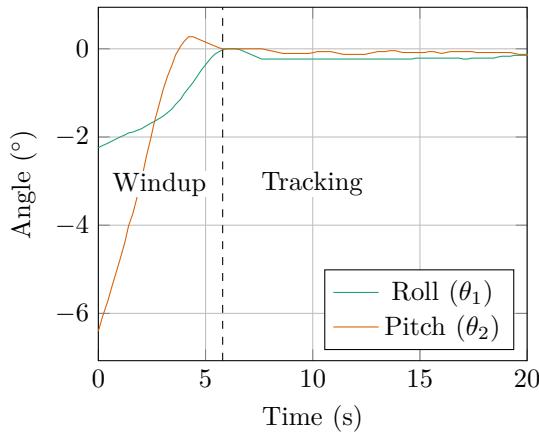


Fig. 22: The initial startup of the mechanism showing the transition between “windup” and “tracking” states.

### C. Actuated universal joint Experiments

1) *Actuated universal joint Angle Tracking:* Figure 23 plots the tracking response of both the simulation and experiments for both version 1 and version 2. Three trajectories were created to test the capabilities of the mechanism. Two were only in one axis of the Universal Joint, and the third was in both axes. Version 1 was tracked between a setpoint range of  $\pm 10^\circ$  for a single axis, and  $\pm 5^\circ$  for two axes. Because version 2 was designed to increase the AUJ angle range, it was tracked between a setpoint range of  $\pm 10^\circ$  for a single axis, and  $\pm 5^\circ$  for two axes.

2) *Active transmission adjustment (ATA):* In version 1, ATA was not possible because of the limited angle range as discussed in section ???. However, an attempt was made to increase the angle range by increasing  $f_{\min}$  in the same fashion. An experiment was set up where the setpoint angle of the AUJ would continually linearly increase, after a short period of acceleration, on the roll ( $\theta_1$ ) axis. The control system monitored the motor angles of each TSA, and halted the experiment when any TSA motor angle was at or below  $2\pi$ , as a safety margin to prevent damage to the mechanism. The AUJ angle at the end of the experiment was recorded, as shown in figure 24. By repeating the experiment with increasing values of  $f_{\min}$ , it was possible to determine how it affected the maximum roll angle. These experiments were able to achieve modest increases of  $4.004^\circ \text{N}^{-1}$  for the positive (upper) limit of the universal joint roll  $\theta_1$ , and  $-6.062^\circ \text{N}^{-1}$  for the negative (lower) limit, within the  $f_{\min}$  interval [3, 3.5]. Attempts to increase the upper bound of this interval resulted in string failure, possibly due to  $l_s(\theta_s)$  exceeding  $l_{\min}$  but it was not possible to conclusively determine this. Similar experiments for the pitch angle resulted in string failure before any TSA fully unwound even at  $f_{\min} = 3$ .

In version 2, the reduced value of  $r$  allowed for ATA. An experiment was set up where instead of trying to find the maximum angle, the roll experiment from section ?? was repeated for increasing values of  $f_{\min}$ . The current from each motor was measured during the experiment, with

TABLE III: Trapezoidal trajectory sequence parameters.

Cycle	Max./Min. Angle [rad]	Max. Velocity [rad s <sup>-1</sup> ]	Acceleration [rad s <sup>-2</sup> ]
1	0.2	0.01	0.00125
2	0.2	0.03	0.02125
3	0.2	0.05	0.04125
4	0.2	0.07	0.06125

the sum of the motor currents predicted to increase as  $f_{\min}$  increases, which would reduce the maximum force any one of the TSA could exert, as explained in section ???. These results can be seen in figure ??, which confirms this prediction, showing the peak and average sum current consumption increasing non-linearly as  $f_{\min}$  increases. Because the TSA are not running at anywhere near maximum velocity due to the damage it may cause to the AUJ components, it is not possible to realise the benefits with this experiment.

3) *The Effect of AUJ Angular Velocity on AUJ Angle Tracking:* To verify the performance of the AUJ at higher angular velocities, the single axis experiments from section V-C1 were repeated with a trapezoidal “chirp” signal, with the maximum angular velocity increasing by  $\omega_0 + (2(n-1)\omega_0)$  each cycle, where  $n$  is the cycle number, and  $\omega_0$  is the initial maximum angular velocity, and the angular acceleration increasing by  $\alpha_0 + (16(n-1)\alpha_0)$  each cycle, where  $\alpha_0$  is the initial angular acceleration. A total of four cycles were performed, with the maximum velocity and acceleration values shown in table III. Figure 25 shows the results of these experiments for both version 1 and version 2, which shows the tracking error increasing marginally as maximum angular velocity and acceleration increases.

4) *The Effect of Follower Mass on AUJ Angle Tracking:* In order to demonstrate the feasibility of a multiple joint system, additional mass was added to the follower segment and the *Pitch & Roll* trajectory was performed to analyze the effect of the additional mass on the control system. This was performed by the modification of the end of the follower segment to allow multiple standard disc weights, each with a mass of 100 g, to be attached. Four experiments were performed, one with no disc weights attached, and three with one, two, and three disc weights attached, adding approximately 100 g each time. 3D printed custom length spacers were used to maintain an approximately equal COM for each of the experiments with disc weights. Table IV lists the four follower mass configurations that were tested, with follower mass and inertia matrix  $m$  and  $I$ , and follower COM z component  $\rho_3$ , along with a figure for each configuration. The follower COM x and y components were set to 0 for these experiments, as they did not vary significantly between experiments, and had a negligible contribution to the dynamics. These figures are based on estimates from a computer aided design (CAD) model developed in SolidWorks®, with known weights of 100 g. Any inertial terms less than  $1 \times 10^{-5}$  were rounded down to 0, which resulted in a diagonal inertia matrix. For

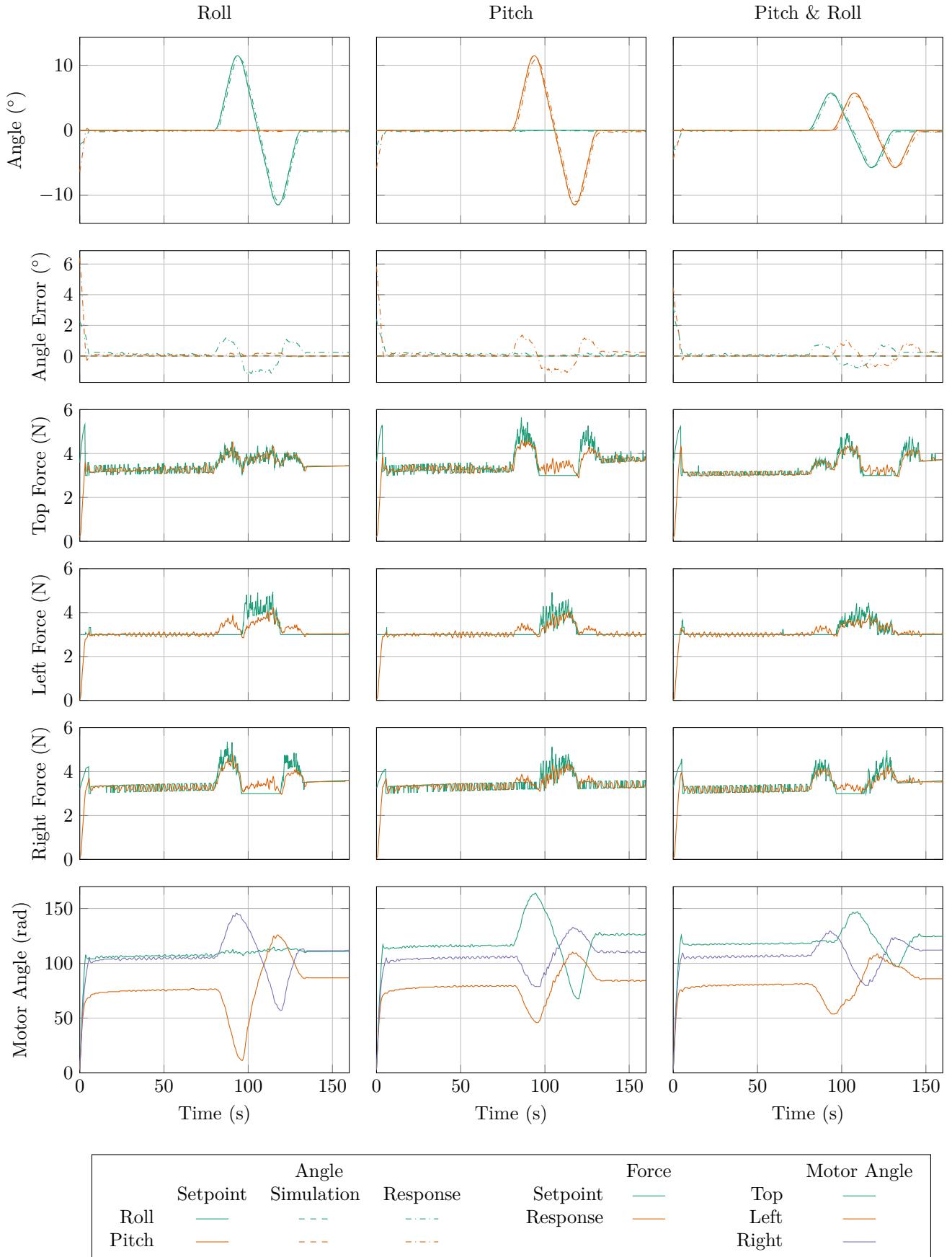


Fig. 23: Plots of the response for three different trajectories, one on only the roll axis  $\theta_1$  (column 1), one on only the pitch axis  $\theta_2$  (column 2), and one on both axes  $\theta_1$  and  $\theta_2$  (column 3). Plots include AUJ orientation, forces at the top, left and right TSA, and the motor positions. Note the simulation error is very small, so the plot cannot be seen on the graph.

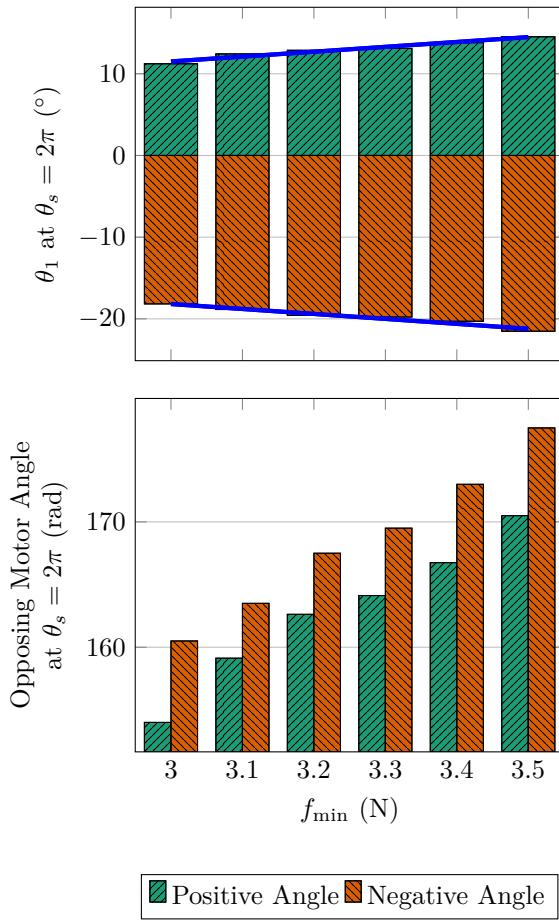


Fig. 24: AUJ roll angle ( $\theta_1$ ) when the smallest TSA motor angle is equal to  $2\pi$ , and motor angle of the “opposing” TSA (the TSA with the largest motor angle) at the same position.

each experiment, the inverse dynamics parameters  $m$  and  $\rho$  were updated so the correct setpoint torque would be calculated.

Figure 26 shows the setpoint and response trajectory for each experiment for both version 1 and version 2. Initially,  $k_p$  was set to the same value as in table I, however the system was unable to reach a steady state in any configuration other than “No Mass”. Reducing  $k_p$  to 1000 in the weighted configurations solved the steady state issue and resulted in an average maximum tracking error of  $0^\circ$  over all configurations with added mass, similar to the result from the initial experiments. However, for the “No Mass” configuration,  $k_p = 1000$  resulted in a very poor tracking response, whereas  $k_p$  from table I resulted in a maximum tracking error of  $0^\circ$ , once again similar to the result from the initial experiments. In future implementations gain scheduling can be employed to select the most optimal  $k_p$  for a given follower mass, that allows for the smallest tracking error while being able to reach a steady state.

## VI. DISCUSSION

### A. TSA Comparison with Leadscrew and Direct Drive

In any practical application of the TSA AUJ, there needs to be some advantages over alternative actuation methods. In this case only “inline” actuation methods will be considered as alternatives, where the actuation system is self contained within the segment, since external actuators would have a clear mass advantage but may not be suitable for an AUJ application. Two metrics can be considered, the maximum tension force  $f_{\max}$ , and the maximum stroke velocity  $\dot{p}_{\max}$ .

1) *Twisted string actuator:* For the TSA metrics, the equations from [8], in particular  $h(\theta)$  and  $k(\theta)$  as used for the state space, which can be used to determine  $f_{\max}$  and  $\dot{p}_{\max}$ . By extracting coefficient  $r_s$  as an input to make  $f(p, r_s)$  and  $\dot{p}(\dot{\theta}, p, r_s)$  the performance of different string thicknesses can be compared for a given unwound length  $l_u$  and  $\tau_{\max}, \dot{\theta}_{\max}$  over the range of the contraction length  $p$ .  $f_{\max}$  is calculated as

$$\begin{aligned} k(\theta) &= l_u - \sqrt{l_u^2 - \theta_s^2 r_s^2} \\ k^{-1}(p) &= \pm \frac{\sqrt{p(2l_u - p)}}{r_s} \\ h^{-1}(\theta) &= \frac{\sqrt{l_u^2 - r_s^2 \theta^2}}{r_s^2 \theta} \end{aligned} \quad (9)$$

$$f(p) = h^{-1}(k^{-1}(p)) = \pm \frac{\sqrt{(l_u - p)^2}}{r_s \sqrt{p(2l_u - p)}}$$

$$f_{\max} = f(p)\tau_{\max},$$

and  $\dot{p}_{\max}$  is calculated as

$$\begin{aligned} k(\dot{\theta}, \theta) &= \frac{\dot{\theta} r_s^2 \theta}{\sqrt{l_u^2 - r_s^2 \theta^2}} \\ \dot{p}(\dot{\theta}, p) &= \dot{k}(\dot{\theta}, k^{-1}(p)) = \pm \frac{\dot{\theta} r_s \sqrt{p(2l_u - p)}}{\sqrt{(l_u - p)^2}} \\ \dot{p}_{\max} &= \dot{p}(\dot{\theta}_{\max}, p). \end{aligned} \quad (10)$$

2) *Leadscrew:* For the leadscrew metrics, the raising torque calculation [35] can be used as the absolute value of  $f_{\max}$ , since the TSA only operates in tension, which can be used to determine the same metrics. The performance of different screw diameters  $d_m$  and leads  $\lambda$  can then be compared for a given  $\tau_{\max}$  and coefficient of friction  $\mu$ .  $\dot{p}_{\max}$  is then calculated by multiplying  $\lambda$  with  $\dot{\theta}_{\max}$ . The performance of different  $\lambda$  can then be compared for a given  $\dot{\theta}_{\max}$ .  $f_{\max}$  is calculated as

$$\begin{aligned} |\tau(f)| &= \frac{d_m f (\lambda + \pi d_m \mu)}{2(\pi d_m - \lambda \mu)} \\ |f(\tau)| &= \frac{2\tau(\pi d_m - \lambda \mu)}{d_m (\lambda + \pi d_m \mu)} \\ f_{\max} &= |f(\tau_{\max})|, \end{aligned} \quad (11)$$

and  $\dot{p}_{\max}$  is calculated as

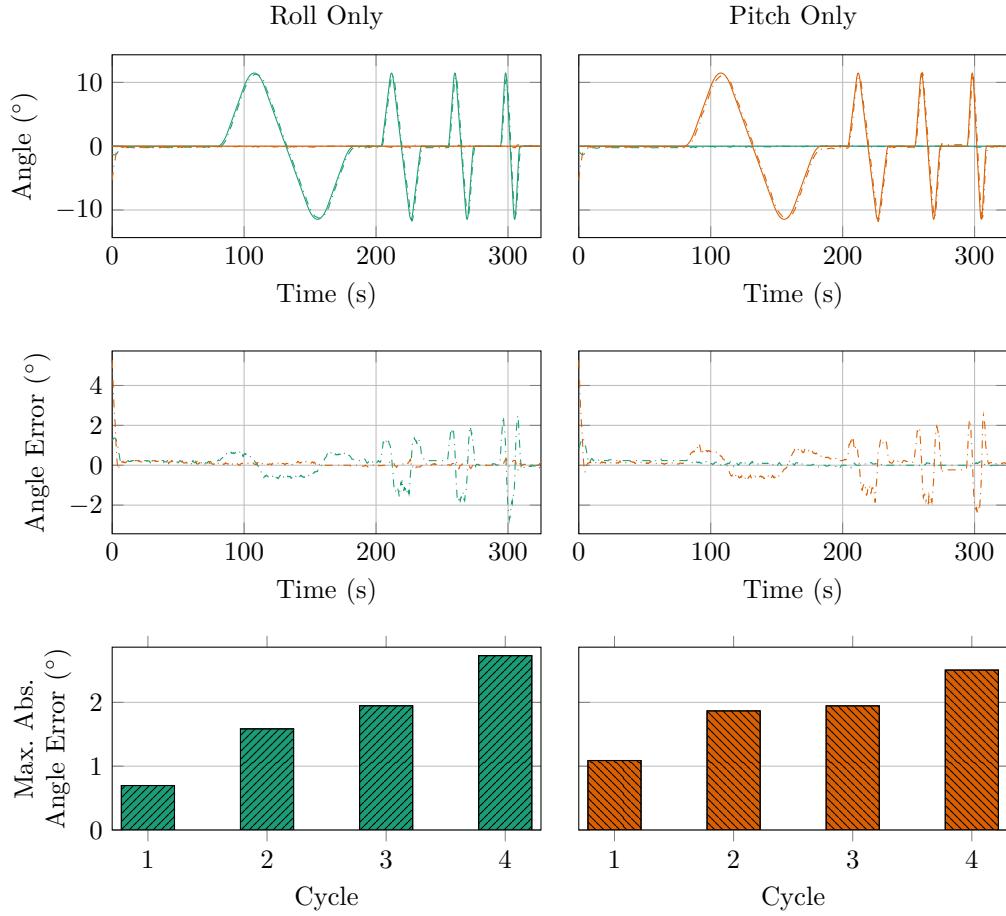


Fig. 25: Results of the trapezoidal velocity trajectory from table III for both AUJ pitch and roll trajectories, including the maximum absolute angle error for each cycle.

TABLE IV: Table of all the follower mass configurations, with the parameters for follower mass  $m$  and follower COM z offset  $\rho_3$ .

Configuration	$m$ [g]	$\rho_3$ [mm]	$I$ [kg m <sup>-2</sup> ]	Image
No Mass	67	6.7	$\begin{bmatrix} 3.3 \times 10^{-5} & 0 & 0 \\ 0 & 3.1 \times 10^{-5} & 0 \\ 0 & 0 & 1.4 \times 10^{-5} \end{bmatrix}$	
+100 g	170	49	$\begin{bmatrix} 1.3 \times 10^{-4} & 0 & 0 \\ 0 & 1.3 \times 10^{-4} & 0 \\ 0 & 0 & 4.6 \times 10^{-5} \end{bmatrix}$	
+200 g	220	50	$\begin{bmatrix} 1.7 \times 10^{-4} & 0 & 0 \\ 0 & 1.7 \times 10^{-4} & 0 \\ 0 & 0 & 7.7 \times 10^{-5} \end{bmatrix}$	
+300 g	320	50	$\begin{bmatrix} 2.0 \times 10^{-4} & 0 & 0 \\ 0 & 2.0 \times 10^{-4} & 0 \\ 0 & 0 & 1.1 \times 10^{-4} \end{bmatrix}$	

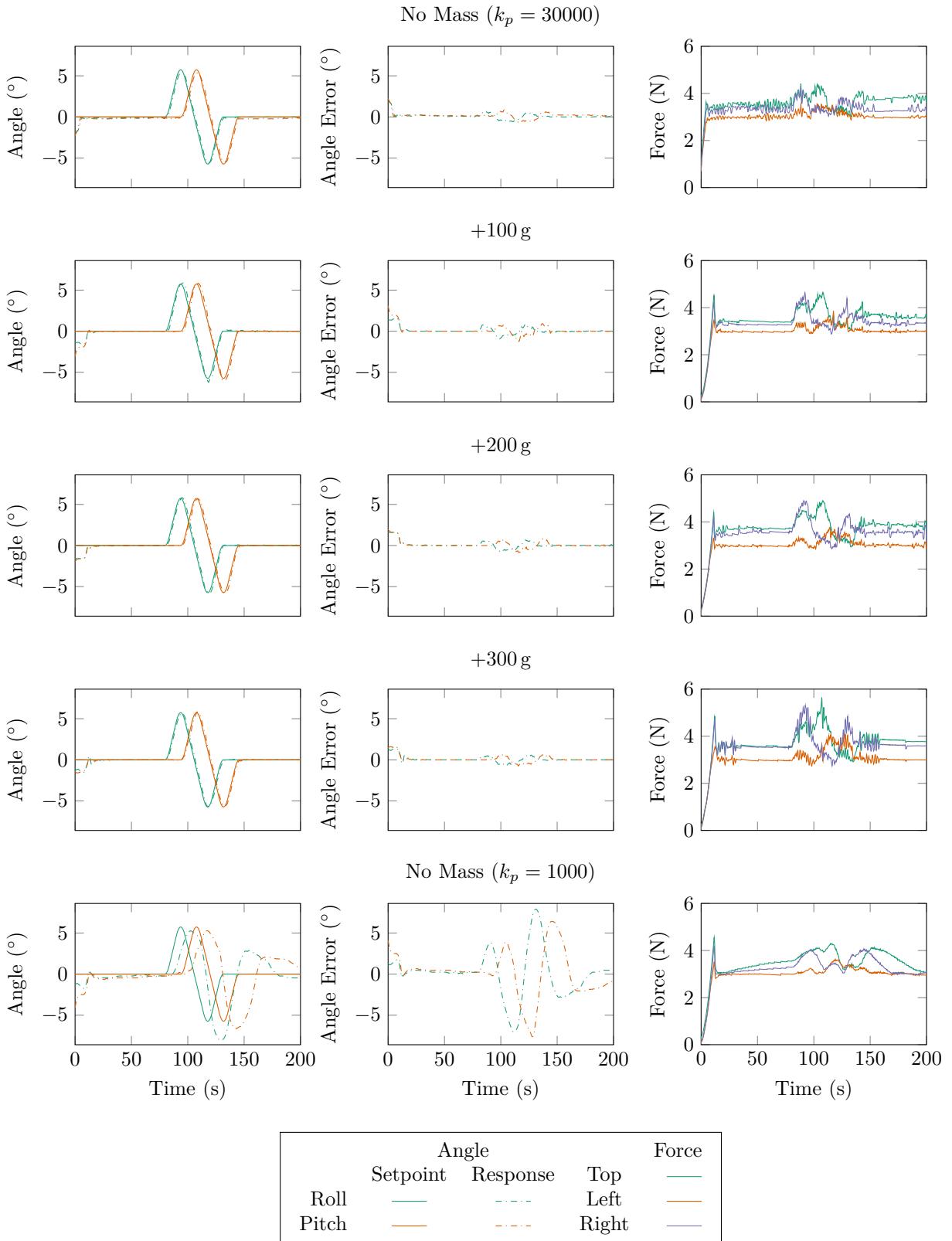


Fig. 26: AUJ pitch and roll tracking from figure 20 with increasing follower mass from table IV, as well as two different values of  $k_p$  for the “No Mass” configuration.

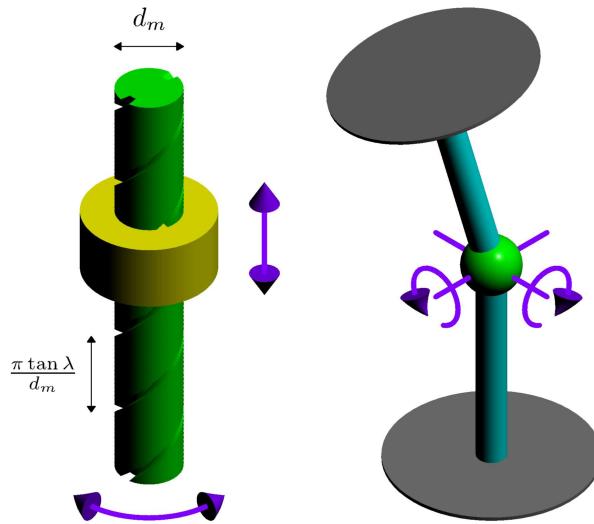


Fig. 27: Abstract diagram of a leadscrew (left) and direct drive system (right). As the screw rotates, the travelling nut moves up and down, creating a linear actuator. The direct drive employs rotary actuators to change the orientation of the AUJ directly.

$$\begin{aligned} \dot{p}(\dot{\theta}) &= \lambda \frac{\dot{\theta}}{2\pi} \\ \dot{p}_{\max} &= \dot{p}(\dot{\theta}_{\max}). \end{aligned} \quad (12)$$

3) *Direct Drive*: The direct drive metrics are trivially calculated using the lever force and tangential velocity that would be generated at the endpoint of a linear actuator able to impart equivalent angular velocity and torque on the universal joint.  $f_{\max}$  is calculated as

$$f_{\max} = \frac{\tau_{\max}}{\sqrt{l_2^2 + r^2}}, \quad (13)$$

and  $\dot{p}_{\max}$  is calculated as

$$\dot{p}_{\max} = \dot{\theta}_{\max} \sqrt{l_2^2 + r^2}. \quad (14)$$

4) *Comparison between TSA and Leadscrew*: As the values for  $\tau_{\max}$  and  $\dot{\theta}_{\max}$  for the TSA depend on  $p$ , but remain constant for the leadscrew, the performance of the TSA is going to be better or worse than a given leadscrew depending on the value  $p$ . Figure 28 compares the TSA configuration using the coefficients from table I against a number of common leadscrew configurations that are practical for the dimensions of the AUJ. The TSA outperforms or underperforms different leadscrew configurations depending on  $p$ . In simpler terms, the performance of the TSA is dependent on the contraction length. The maximum linear velocity increases with the contraction length, and the maximum tension force decreases with contraction length, both in a nonlinear fashion.

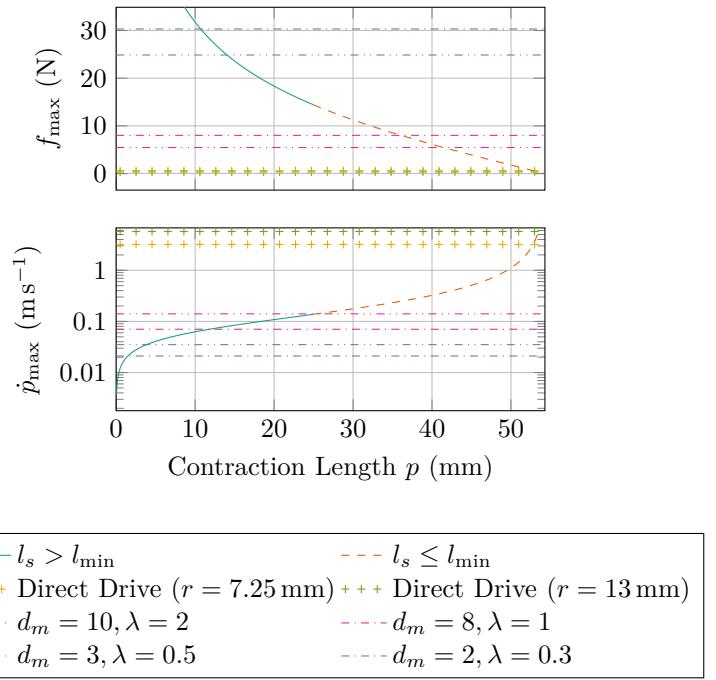


Fig. 28: Performance comparison of the TSA configuration using coefficients from table I to various leadscrew configurations with different  $d_m$  and  $\lambda$ , and the direct drive, where  $\mu = 0.1$  for the leadscrews.

### B. Alternative Control Algorithm

Alternative control algorithms that are less or not dependant on the dynamic properties of the follower segments (mass and inertia) will also be considered, in order to improve the robustness of the control system to external forces.

### C. Multiple Segments

Development of a system comprised of multiple segments is the eventual goal of the research, to demonstrate its suitability for applications such as mobile snake robots or snake-arm robots. This can use a embedded controller for each segment, controlled by a primary controller for individual joint control or inverse kinematics, as in figure 29.

## VII. CONCLUSION

This research has successfully demonstrated force based control of the orientation of a universal joint using TSA in an antagonistic triad configuration, at various velocities and follower loads. It has also compared the performance of the system to alternative actuation methods, and found that its non-linear nature makes it difficult to directly compare to similar linear actuators in the same application. Finally, it has examined the limitations of the current design, and has proposed design modifications to improve range and accuracy.

In this research, a light, compact and adaptable method for electrically actuating a universal joint using inline

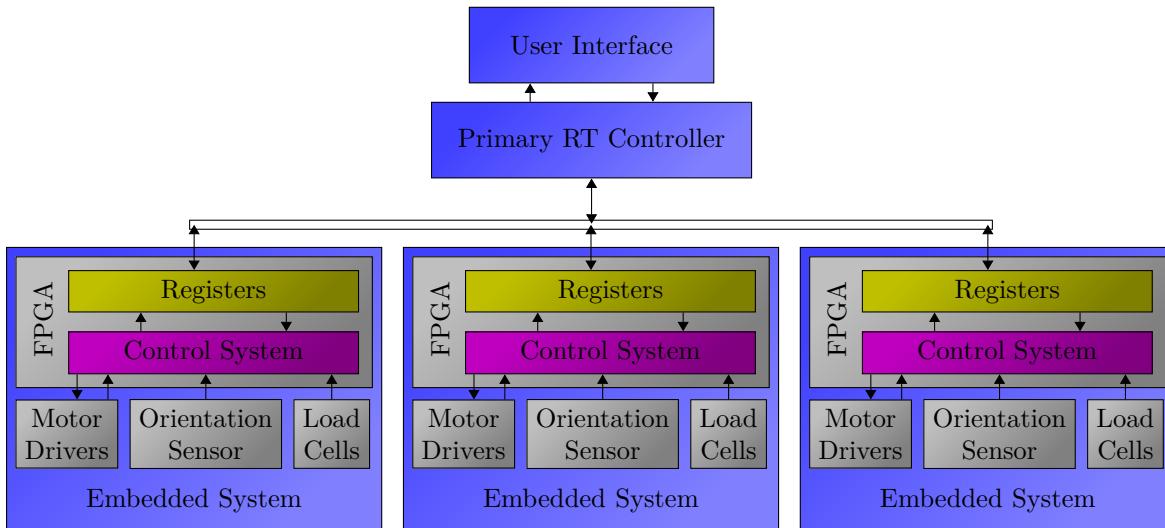


Fig. 29: Proposed system architecture for a future multi-segment system. Each segment has an embedded FPGA controller programmed with the cascaded control loop in section II. The FPGA interfaces with the load cells for each TSA, the orientation sensor for the AUJ, and the drivers for the TSA motors (the FPGA may include some motor control for a more compact design). A primary real-time controller then uses a common control bus to interface with the embedded controllers, reading and writing FPGA registers to issue motion commands and get status updates.

actuators has been conceptualised, tested and evaluated using various performance characteristics. This method has numerous potential applications in mobile robotics and robotic manipulators where a soft-rigid mechanism is required, such as mobile snake robots, snake-arm robots or robotic tails.

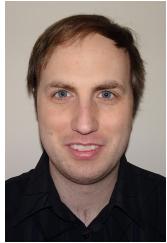
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