

### CEA analysis performed on Sat 09-Nov-2024 01:27:28

# Problem Type: "Rocket" (Infinite Area Combustor)

prob case=\_\_\_\_\_1032 ro equilibrium

# Pressure (10 values):

p,atm= 9.869, 19.738, 29.607, 39.476, 49.345, 59.214, 69.083, 78.952, 88.821, 98.69

# Chamber/Exit Pressure Ratio (1 value):

pi/p= 99

# Oxidizer/Fuel Wt. ratio (1 value):

o/f= 3.0

# You selected the following fuels and oxidizers:

reac

fuel AL wt%= 11.0000

fuel C2H5OH wt%= 12.2000

fuel C3H8O,2propanol wt%= 4.1000

fuel HNC0 wt%= 2.7000

fuel NH4CLO4(I) wt%= 70.0000

oxid NH4CLO4(I) wt%=100.0000

# You selected these options for output:

# short version of output

output short

# Proportions of any products will be expressed as Mass Fractions.

output massf

# Heat will be expressed as siunits

output siunits

# Input prepared by this script:/var/www/sites/cearun.grc.nasa.gov/cgi-bin/CEARUN/prepareInputFile.cgi

### IMPORTANT: The following line is the end of your CEA input file!

end