

## Cosmic Dawn Intensity Mapper: Spacecraft and Mission Design for a Probe-Class Space Telescope

Philip Linden<sup>1,†</sup>, Michael Zemcov<sup>2</sup>

<sup>1</sup>*Department of Mechanical Engineering, Kate Gleason College of Engineering, Rochester Institute of Technology, Rochester, NY 14623, USA, [pjl7651@rit.edu](mailto:pjl7651@rit.edu)*

<sup>2</sup>*Center for Detectors, School of Physics and Astronomy, Rochester Institute of Technology, Rochester, NY 14623, USA, [zemcov@cfid.rit.edu](mailto:zemcov@cfid.rit.edu)*

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Cosmic Dawn Intensity Mapper (CDIM) is a Probe-class near-IR space telescope with the scientific goal of conducting large spectro-imaging surveys over a five-year period in the 2020 Decadal. A high-level system architecture was designed to identify key features and technologies aboard the CDIM spacecraft in preparation for more detailed studies such as a Team-X session at NASA Jet Propulsion Laboratory.

*Keywords:* spacecraft, telescope, system, cryogenic, infrared, design.

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<sup>†</sup>Corresponding author.

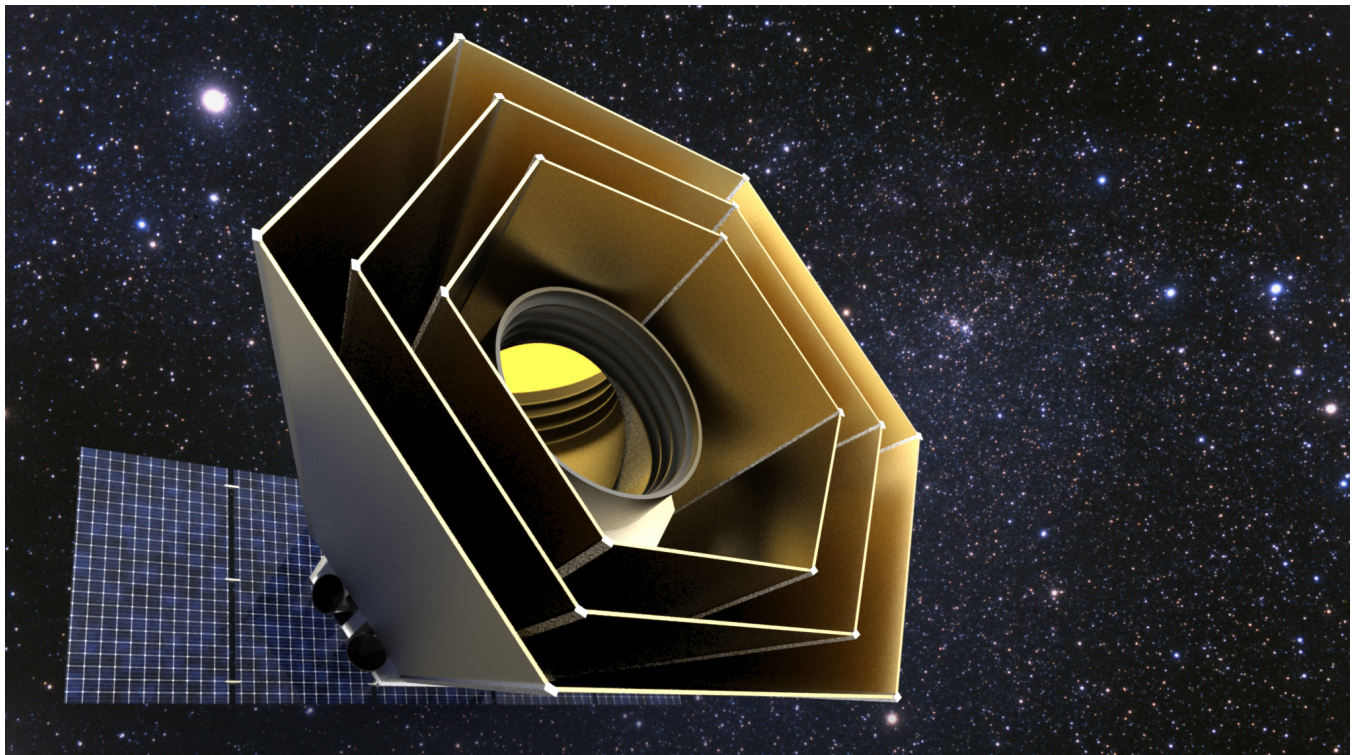


Fig. 1. An artistic rendition of the Cosmic Dawn Intensity Mapper stationed at Sun-Earth L<sub>2</sub>.

## 1. Introduction

Observing the behavior and characteristics of the earliest stars and galaxies is fundamental to understanding the physics that led to their formation and evolution. Breakthrough discoveries in understanding the physics of the epoch of reionization are anticipated in the 2020–2030 decade thanks to WFIRST and JWST. However, JWST’s capability will be limited to only several cosmological deep fields. Although WFIRST will be capable of wide area surveys, its spectroscopy is limited to  $2\mu\text{m}$  and thus limits the selection of galaxies it is able to observe. Neither JWST nor WFIRST provide a complete understanding of the epoch of reionization, specifically in terms of answering the questions of when and how the cosmic dawn came to be. This area of research is a prime candidate for a Probe class mission optimized to study reionization.

Probe class missions occupy a role on a larger scale than Discovery missions, such as Kepler and Dawn, but not as vast as Flagship missions such as JWST (Wiseman *et al.*, 2015). Such missions are intended to be PI-led scientific investigations rather than general observatories, and have a firm \$1B cap.

Cosmic Dawn Intensity Mapper (CDIM) is a Probe class infrared telescope capable of three-dimensional spectro-imaging observations over the wavelength range of  $0.75$  to  $7.5\mu\text{m}$ , at a spectral resolving power  $\Delta\lambda/\lambda$  of 500. CDIM has a  $10\text{deg}^2$  instantaneous field of view (FoV) utilizing linear variable filters (LVFs) atop a focal plane of thirty-six  $2048 \times 2048$  detectors. The survey strategy using spacecraft operations following a shift and stare mode will result in 1360 independent narrow-band spectral images of the sky on a given location. Surveys could span from  $25\text{deg}^2$  up to  $1000\text{deg}^2$  over a five year lifetime in an orbit about Sun-Earth Lagrange point  $L_2$ .

With these instrument requirements, CDIM is optimized to search for the first cosmic sources of dust and evidence of the very earliest stellar populations, bridging the gaps in JWST and WFIRST cosmic dawn surveys and exceeding them in capability.

CDIM is notionally a  $1.5\text{m}$  aperture telescope, passively cooled to  $45\text{K}$ , with a  $6 \times 6$  detector array that utilizes linear variable filters (LVFs) and actively cooled to  $35\text{K}$  (Cooray *et al.*, 2016).

## 2. Optical Telescope Array

Preliminary explorations indicate that a  $1.5\text{m}$  aperture off-axis primary mirror cooled to  $45\text{K}$  is required to meet CDIM’s spectro-imaging requirements (Cooray *et al.*, 2016). The primary mirror is notionally assumed to be constructed from light-weighted Corning (ultra-low expansion) silica-titania glass with a honeycomb core and a gold-deposition surface coating. “Back-of-the-envelope” calculations estimate the primary mirror’s mass to be roughly  $200\text{kg}$ .

Multiple detectors satisfy CDIM’s spatial resolution, wavelength range, and sensitivity requirements. These detectors range in TRL, but all are sufficiently developed to be considered for the 2020 Decadal and have been demonstrated on missions such as NEOCam, SPHEREx, and JWST.

Teledyne H2RG-18 HyViSI detectors offer a  $2048 \times 2048$  pixel array format at a pixel pitch of  $18\mu\text{m}$  (Bai *et al.*, 2008). CDIM will utilize a  $6 \times 6$  H2RG array. Each detector nominally dissipates less than  $4\text{mW}$ , for a total power dissipation of less than  $150\text{mW}$  for the array.

CDIM optics, instruments, and focal plane will be housed in a light-tight box. The optical telescope assembly (OTA) as a whole is estimated to have a mass of  $x\text{kg}$ .

## 3. Thermal Regulation

CDIM employs both passive and active thermal regulation systems. By using passive radiators in tandem with an active cryocooler, the static OTA heat load can be dissipated by the lightweight radiator and a smaller cryocooler may be used to only cool the detector array rather than the whole OTA mass plus focal plane assembly.

The OTA is cooled to  $45\text{K}$  passively to reduce background photon load in the near-IR. Passive thermal regulation is maintained using a V-groove radiator, which bounces radiative heat into the  $3\text{K}$  background of space. V-groove radiators have been demonstrated in passive cryogenic radiators up to  $4\text{K}$  with Planck, SPIRIT, and Spitzer (warm mission). In order to achieve passive cooling from a baseline temperature of  $300\text{K}$  at Sun-Earth Lagrange point  $L_2$ , a  $x$ -stage V-groove radiator with an area of  $x\text{m}^2$  is required.

The CDIM detector array is actively cooled to  $35\text{K}$  to reduce thermal noise. Stirling-cycle mechanical refrigerators are low-vibration, high-reliability, and lightweight active cryocooling systems that

Table 1. Critical design requirements for the CDIM spacecraft following the format suggested by Wertz *et al.*

Spacecraft Design Driver	Impact	Target
Cost	Quality of parts	less than \$1 B
Mass	Launch vehicle	less than 1000 kg
Temperature (OTA)	Cryocooler, radiator	45 K
Temperature (Detector)	Cryocooler, radiator	35 K
Pointing Requirements	Attitude control sys	0.5 arcsec
Lifetime	Redundancy, RCS propellant	5 years
Orbit	Solar array, thermal management, launch vehicle, telemetry	Sun-Earth L <sub>2</sub>

have significant heritage in space applications. One candidate system is Raytheon’s PSC 1-stage Stirling cryocooler, capable of cooling a 1.2 W parasitic heat load to 35 K. This cryocooler is 18.6 kg and requires 88 W of input power (Donabedian *et al.*, 2003).

#### 4. Attitude Determination and Control

To conduct a survey, the spacecraft must first understand its orientation and then act to align itself with a given area of the sky. Redundant systems of varying levels of fidelity are included to allow CDIM to operate in different power states. Low-fidelity attitude determination sensors such as sun sensors are cheap, accurate to less than one degree, and lightweight.

High-fidelity attitude determination is conducted by star tracking cameras. Star trackers identify constellations in their field of view to determine the spacecraft’s heading to within 1 arcsecond.

In a heliocentric orbit, the primary disturbance to the spacecraft’s heading is solar radiation pressure (SRP). At L<sub>2</sub>, solar radiation pressure presents itself as a constant torque on the spacecraft on the order of  $10^{-5}$  N m.

Cold-gas or Hydrazine thrusters will be used for orbit station-keeping as well as momentum management. A desired heading is maintained by the spacecraft using a 3-axis zero-momentum inertial system, whereby the error in heading due to SRP is cancelled out by spinning up or slowing down reaction wheels. Reaction wheel systems are capable of torques ranging from .01 to 1 N m and store 0.4 to 3000 N m s of practical momentum (Wertz *et al.*, 2015). Power consumption varies with reaction wheel speed, with a maximum estimate of roughly 100 W. Since SRP is constant, after some time the inertial attitude control will become saturated. Desaturation is managed by engaging station-keeping thrusters for short periods of time.

#### 5. Flight Computer

#### 6. Telemetry

Typically for high-Earth and deep-space missions, the X-Band Space Science frequency band is used for uplink and downlink between the spacecraft and Ground Stations. Thus, high-gain antennas are best suited for both links (Wertz *et al.*, 2015).

Table 2. For redundancy, CDIM is outfitted with multiple communication modes. Typical data transfer rates are outlined for each mode (Wertz *et al.*, 2015).

Mode	Uplink	Downlink
Emergency	7.8 bps	5 to 10 bps
Engineering data	15.6 to 2000 kbps	Up to 10 bps
Science data	15.6 to 2000 kbps	10 Mbps or more

#### 7. Power

All power generation will come from an array of photovoltaic cells facing the sun. In order to survey the entire sky, the cells must be able to adjust to account for different incident angles to the sun. The array must deploy after the launch and orbital insertion phases of the mission.

Based on a rough power budget and the spacecraft’s position at L<sub>2</sub>, the array must be  $x\text{m}^2$  in area to sustain operation. The dark side of the array acts as a radiator to contribute to the thermal regulation of the spacecraft bus.

#### 8. Structure

The spacecraft bus houses all non-instrumentation systems including the cryocooler, ADCS, and power modules. It would have approximately some dimensions and be made of standard materials. The bus will also include hard points for integration with the launch vehicle.

Table 3. Available launch vehicles and their capabilities to send payloads to L<sub>2</sub> (Rioux, 2016; SpaceLaunchReport).

Vehicle	Payload to L <sub>2</sub>	Fairing size	Cost
Falcon 9 v1.1	2900 kg	5.2 × 13.1 m	\$62M
Atlas V 551	6100 kg	4.2 × 10.0 m 5.1 × 11.0 m	\$153M
Ariane V	6600 kg	5.4 × 12.7 m 5.4 × 13.8 m 5.4 × 17.0 m	\$165M \$220M
Delta IV Heavy	9800 kg	5.0 × 14.3 m 5.0 × 19.1 m	\$375M

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9. Mission Profile

9.1. Launch Vehicle

9.2. Operations

9.3. End of Life

10. Budgets & Costs

11. Conclusion

Acknowledgments

Thanks.

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