We wish to design a controller for the elevator such that the aircraft is able to:

- \bullet maintain its elevation even in the presence of disturbances such as wind and sudden
- increase altitude of 1 m in 10s (the settling time to a unit step of the closed loop system should be < 10 s),

$$P(S) = \frac{0.4}{5^3 + 0.15^2 + 205 + 0.5} = \frac{0.4}{(s + 0.025)(s + 0.0375 + 4.47)(s + 0.0375 - 4.47)}$$

Red[1:
$$M_{\Gamma} = \frac{\alpha^2 + \beta^2}{2 \times \beta}$$

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$$2 \alpha \beta M_{\Gamma} = \alpha^{2} + \beta^{2}$$

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$$\alpha^{2} - 2 \cdot l.5 \alpha \beta + \beta^{2} = 0$$

$$\alpha^{2} - 3 \alpha \beta + \beta^{2} = 0$$

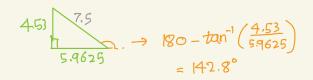
$$\alpha = 3 \beta \pm \sqrt{(3\beta)^{2} - 4 \beta^{2}}$$

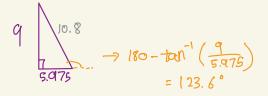
$$2 \alpha = 3 \beta \pm \sqrt{5 \beta^{2}}$$

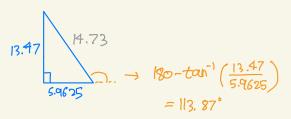
$$2 \alpha = 3 \beta \pm \sqrt{5 \beta^{2}}$$

$$20 = 3\beta \pm \sqrt{5}\beta^{2}$$

$$x = \frac{3\beta \pm \sqrt{5}\beta}{2}$$





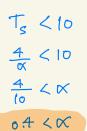


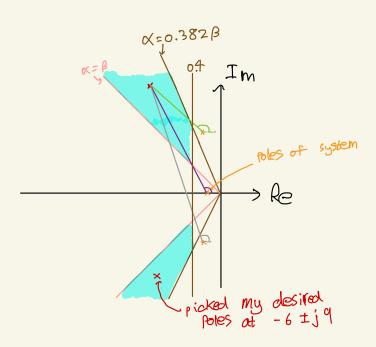
$$\frac{0.4}{7.5/142.8^{\circ} \cdot 10.8 \cdot 123.6^{\circ} \cdot 14.73/113.87^{\circ}} \quad k_{P} \frac{M_{Z} \phi_{Z}}{M_{P} \phi_{P}} = 1 / -180^{\circ}$$

$$-142.8 - 123.6 - 113.87 + \phi_2 - \phi_P = -180$$

$$\phi_{z} - \phi_{p} = 200.27$$

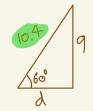
 $\phi_2 - \phi_p = 200.27$ Must use compensator, but one is not enough should aim for \$z < 60° ... use 4 compensators





$$\frac{200.27}{4} = 50.0675$$

$$\phi_{z} - \phi_{p} = 50^{\circ}$$
Pick $\phi_{z} = 60 \rightarrow : \phi_{p} = 10$



$$tan (60) = \frac{9}{d}$$

 $d = 5.2$

$$tan(10) = \frac{9}{d}$$

$$d = 51$$

$$place pole at
$$6+51 = -57$$$$

Final • 1835 750
$$\frac{(5+11.2)^4}{(5+57)^4}$$

