

Calculate the ideal chamber pressure for a rocket motor having a  $K_n$  of 180

$$K_n = \frac{A_B}{A_*} \xrightarrow{\text{burn area}} = 180 \quad \xleftarrow{\text{throat area}}$$

Steady state chamber pressure =  $P_0$

$$P_0 = \left[ \frac{A_B}{A_*} \frac{\alpha P_p}{\sqrt{\frac{\kappa}{RT_0} \left( \frac{2}{\kappa+1} \right)^{\frac{\kappa+1}{\kappa-1}}} \right]^{\frac{1}{1-n}}$$

$$P_0 = \left[ 180 \cdot \frac{\alpha P_p}{\sqrt{\frac{\kappa}{RT_0} \left( \frac{2}{\kappa+1} \right)^{\frac{\kappa+1}{\kappa-1}}} \right]^{\frac{1}{1-n}}$$

for KNSU,  $\kappa = 1.33$

$$T_0 = 1720 \text{ K}$$

$$M = 41.98 \text{ kg/kmol}$$

$$R' = 8314 \text{ N-m/kmol-K} \quad \text{Universal gas constant}$$

$$R = \frac{R'}{M} \Rightarrow R = 198 \text{ Nm/kg-K}$$

wrong by St. Robert's factor, burn rate can be expressed  
in terms of chamber pressure

$$r = \alpha P_0^n \quad \begin{cases} \alpha: \text{burn rate coefficient} \\ n: \text{constant} \end{cases}$$

by chemistry of KNSU:

$$\alpha = 0.0665 \text{ in/sec - psi}^n \quad P_p$$

$$n = 0.319$$

$$\alpha = 0.0665 \frac{\text{in}}{\text{sec psi}^n} \cdot \frac{1 \text{ m}}{39.37 \text{ in}} \cdot \frac{1 \text{ sec psi}}{(6895)^{0.319} \text{ Pa}^n} = 0.000$$

\* Must convert to SI units

Plug in, get:

$$P_0 = 4029000 \text{ Pa} = 4.03 \text{ MPa} = 58^n \text{ psi}$$

Calculating ideal chamber pressure for APCP  
 Propellant) with a  $K_n$  of 180 - 250  
 (assume 215)

$$K = 1.21 \text{ (ratio of specific heats)}$$

$$T_0 = 3500^\circ K \text{ (adiabatic flame temp)}$$

$$M = 23.67 \text{ g/mol}$$

$$R' = 8314 \text{ N-m/kgmol-K}$$

$$R = \frac{R'}{M} = \frac{8314}{23.67} = 326 \text{ N-m/g-K}$$

$$\rho_p = 1680 \text{ g/m}^3 \text{ (propellant density)}$$

$$a = 3.5130541 \times 10^{-5}, \text{ (burn rate coefficient)}$$

$$n = 0.3273 \text{ (burn rate exponent)}$$

$$P_0 = \left[ \frac{A_D}{A_*} \frac{\alpha_{pp}}{\sqrt{\frac{K}{RT_0} \left( \frac{2}{K+1} \right)^{\frac{K+1}{K-1}}}} \right]^{\frac{1}{1-n}}$$

Calc one component at a time

$$\frac{A_D}{A_*} = K_n = 215$$

$$\frac{K}{RT_0} = 1.060473269 \times 10^{-6}$$

$$\frac{2}{K+1} = 0.9049773756$$

$$\frac{K+1}{K-1} = 10.52380952$$

$$\frac{1}{1-n} = 1.486546752$$

$$\alpha_{pp} = 0.0590865089$$

$$P_0 = \left[ 215 \cdot \frac{0.0590865089}{\sqrt{1.06 \cdot (0.90)^{10.52}}} \right]^{1.486}$$

12.2

$$P_0 = 2635775.616 \text{ Pa}$$

$$P_0 = 382 \text{ psi} = 26.4 \text{ atm}$$

Calculating optimum expansion ratio

$$\frac{A^*}{A_e} = \left( \frac{k+1}{\gamma} \right)^{\frac{1}{k-1}} \left( \frac{P_e}{P_0} \right)^{\frac{1}{k}} \sqrt{\left( \frac{k+1}{k-1} \right) \left[ 1 - \left( \frac{P_e}{P_0} \right)^{\frac{k-1}{k}} \right]}$$

Solve everything involving involving  $k$

$$\frac{k+1}{\gamma} = 1.105$$

$$\frac{1}{k-1} = 4.7619$$

$$\frac{1}{k} = 0.826$$

$$\frac{k+1}{k-1} = 10.5238$$

$$\frac{k-1}{k} = 0.1735$$

$$\frac{P_e}{P_0} = \frac{1}{26} = 0.0384$$

$$\frac{A^*}{A_e} = (1.105) \quad 4.7619 \quad (0.0384) \quad 0.826 \quad \sqrt{(10.52)(1 - (0.03)^{0.17})}$$

$$\frac{A^*}{A_e} = 0.232201$$

$$\frac{A_e}{A^*} = 4.30$$

Optimal expansion ratio