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VOLUME 3

LEVEL II

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EFFECT OF VARIANCES AND MANUFACTURING TOLERANCES ON THE DESIGN STRENGTH AND LIFE OF MECHANICALLY FASTENED COMPOSITE JOINTS

VOLUME 3 - BOLTED JOINT STRESS FIELD MODEL (BJSFM) COMPUTER PROGRAM USER's MANUAL

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This technical report has been reviewed and is approved for publication.

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20. ABSTRACT (Continue on reverse side if necessary and identify by block number)

The subject of this program was structural evaluation of mechanically fastened composite joints. Program objectives were threefold: (1) development and verification by test of improved static strength methodology, (2) experimental evaluation of the effects of manufacturing anomalies on joint static strength, and (3) experimental evaluation of joint fatigue life.

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Program activities to accomplish these objectives were organized under five tasks. Under Task 1 - Literature Survey, a survey was performed to determine the state-of-the-art in design and analysis of bolted composite joints. Experimental evaluations of joint static strength were performed under Tasks 2 and 3. In Task 2 - Evaluation of Joint Design Variables, strength data were obtained through an experimental program to evaluate the effects of twelve joint design variables. In Task 3 - Evaluation of Manufacturing and Service Anomalies, effects of seven anomalies on joint strength were evaluated experimentally and compared with Task 2 strength data. Bolted composite joint durability was evaluated under Task 4 - Evaluation of Critical Joint Design Variables on Fatigue Life. Seven critical design variables or manufacturing anomalies were identified based on Task 2 and 3 strength data. Under Task 5 - Final Analyses and Correlation, required data reduction, methodology development and correlation, and necessary documentation were performed.

This report documents all program activities performed under Tasks 2, 3, 4 and 5. Activities performed under Task 1 - Literature Survey, were previously reported on AFFDL-TR-78-179. Static strength methodology and evaluations of joint static and fatigue test data are reported. Analytic studies complement methodology development and illustrate: the need for detailed stress analysis, the utility of the developed "Bolted Joint Stress Field Model" (BJSFM) procedure, and define model limitations. For static strength data, correlations with analytic predictions are included. Data trends in all cases are discussed relative to joint strength and failure mode. For joint fatigue studies, data trends are discussed relative to life, hole elongation, and failure mode behavior.

This final report is organized in the following three volumes:

Volume 1 - Methodology Development and Data Evaluation
Volume 2 - Test Data, Equipment and Procedures
Volume 3 - Bolted Joint Stress Field Model (BJSFM) Computer Program User's Manual

FOREWORD

The work reported herein was performed by the McDonnell Aircraft Company (MCAIR) of the McDonnell Douglas Corporation (MDC), St. Louis, Missouri, under Air Force Contract F33615-77-C-3140, for the Flight Dynamics Laboratory, Wright-Patterson Air Force Base, Ohio. This effort was conducted under Project No. 2401 "Structural Mechanics", Task 240101 " Structural Integrity for Military Aerospace Vehicles", Work Unit 24010110 "Effect of Variances and Manufacturing Tolerances on the Design Strength and Life of Mechanically Fastened Composite Joints". The Air Force Project Engineer at contract go-ahead was Mr. Roger J. Aschenbrenner (AFWAL/FIBEC); in December 1979, Capt. Robert L. Gallo (AFWAL/FIBEC) assumed this assignment. The work described was conducted during the period 15 February 1978 through 15 April 1981.

Program Manager was Mr. Ramon A. Garrett, Branch Chief Technology, MCAIR Structural Research Department. Principal Investigator was Mr. Samual P. Garbo, MCAIR Structural Research Department.

TABLE OF CONTENTS

Section		Page
I	INTRODUCTION	1
II	PROGRAM DESCRIPTION	2
III	USER'S INSTRUCTIONS	6
IV	OUTPUT OPTIONS	11
V	PROGRAM LIMITATIONS	15
VI	EXAMPLE PROBLEMS	16
VII	PROGRAM LISTING	20

LIST OF ILLUSTRATIONS

Figure No.		Page
1	General Load Conditions Analyzed by BJSFM	3
2	Bolted Joint Stress Field Model - BJSFM (Computer Program Flow Chart)	3
3	Superposition of Linear - Elastic Stress Solutions	4
4	Assumed Cosine Bolt Load Distribution	4
5	Superposition of Infinite Plate Results Approximates a Finite Width Joint	5
6	Program Nomenclature	7
7	Example Input Instructions	10
8	Laminate Stress Distribution Output Data Option	12
9	Ply-by-Ply Strain Distribution Output Data Option	13
10	Loaded Hole Case	17
11	Isotropic Unloaded Hole	18
12	General Loading Condition	19

SECTION I

INTRODUCTION

One objective of this program was to develop a static strength methodology for mechanically-fastened composite joints. This volume documents user-options and instructions for a computer program to analyze the effects of stress concentrations on laminate strength. Entitled "Bolted Joint Stress Field Model" (BJSFM), it computes stress distributions on a lamina or laminate basis for unloaded or loaded (bolt bearing) holes in isotropic or anisotropic materials. Failure predictions based on lamina properties and one of several failure criteria are possible. This volume describes the formulation and input data requirements and output options. Sample problems and a computer program listing are also included.

SECTION II

PROGRAM DESCRIPTION

The Bolted Joint Stress Field Model has been developed to facilitate strength analysis of isotropic or anisotropic materials at individual fastener holes. Static strength of an anisotropic laminate with a fastener hole is predicted using a closed-form analytic approach based on (1) elastic anisotropic theory of elasticity, (2) lamination plate theory and (3) one of several optional failure hypotheses. The program has capability to handle strength and stiffness anisotropy, general in-plane loadings, as shown in Figure 1, multi-material (hybrid) laminates and arbitrary hole (bolt) sizes. BJSFM modular substructuring is illustrated in Figure 2. Input data required are: lamina mechanical properties, in-plane loadings, hole geometry, and hole loading. Options are available which provide computation results after each program block.

The stress field calculations are based on two-dimensional anisotropic theory of elasticity solutions for a homogeneous, anisotropic infinite plate. Laminate stress distributions around an unloaded or loaded (bolt bearing) hole are calculated using plane stress assumptions. Laminate stress and strain distributions for combined bearing and bypass loads are obtained using the principle of superposition (Figure 3). Fastener bearing is idealized as a cosine radial stress distribution (Figure 4). Finite width corrections for loaded holes are based on superposition of infinite plate results as shown in Figure 5. Infinite plate solutions are exact while corrections for finite width joints are approximate and most accurate for width-to-diameter ratios greater than four.

Laminate strains are calculated using material compliance relations. Laminate compliance coefficients are determined using lamination plate theory with unidirectional (lamina) elastic constants, lamina orientations and thicknesses. Strains for individual plies along lamina principal material axes are calculated using coordinate transformations. The solution is strictly valid only for homogeneous media; however, it has been assumed valid for mid-plane symmetric laminates.

Laminate failure is predicted by comparing elastic stress distributions with any of five material failure criteria on a ply-by-ply basis. Failure can be assessed at any location in the field of the plate.

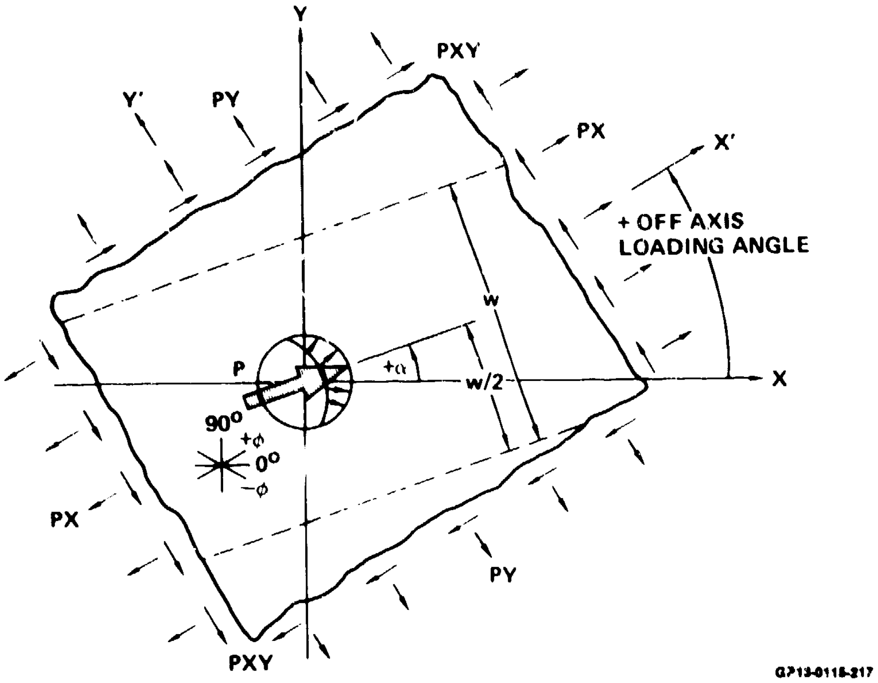

Figure 1. General Load Conditions Analyzed Using BJSFM

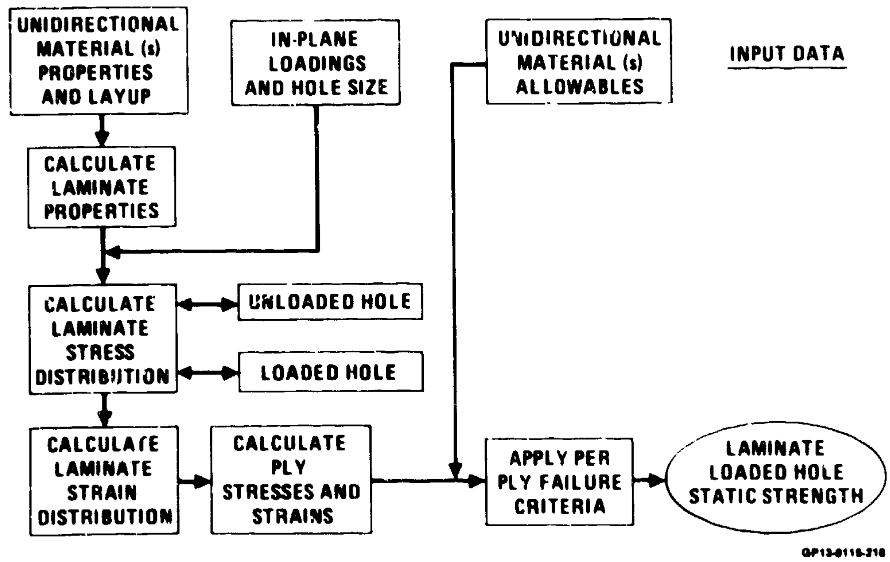

Figure 2. Bolted Joint Stress Field Model - BJSFM
Computer Program Flow Chart

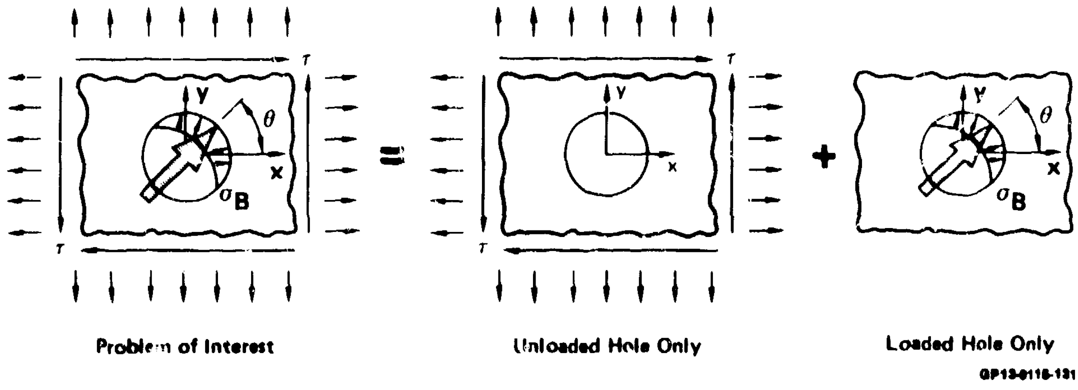

Figure 3. Superposition of Linear-Elastic Stress Solutions

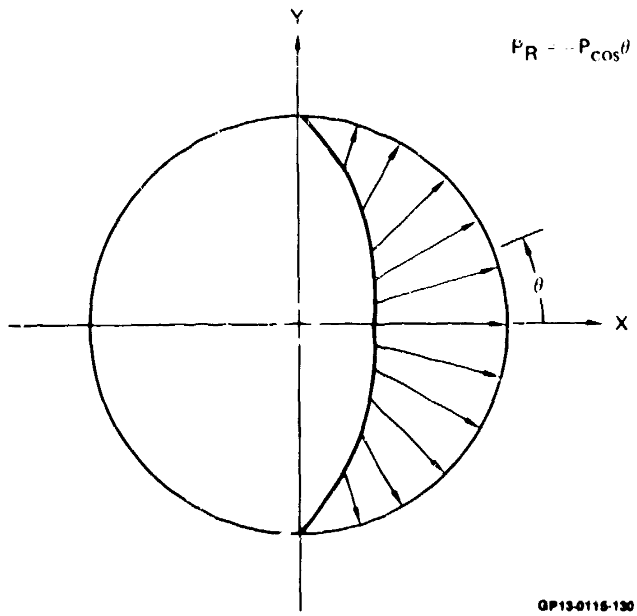

Figure 4. Assumed Cosine Bolt-Load Distribution

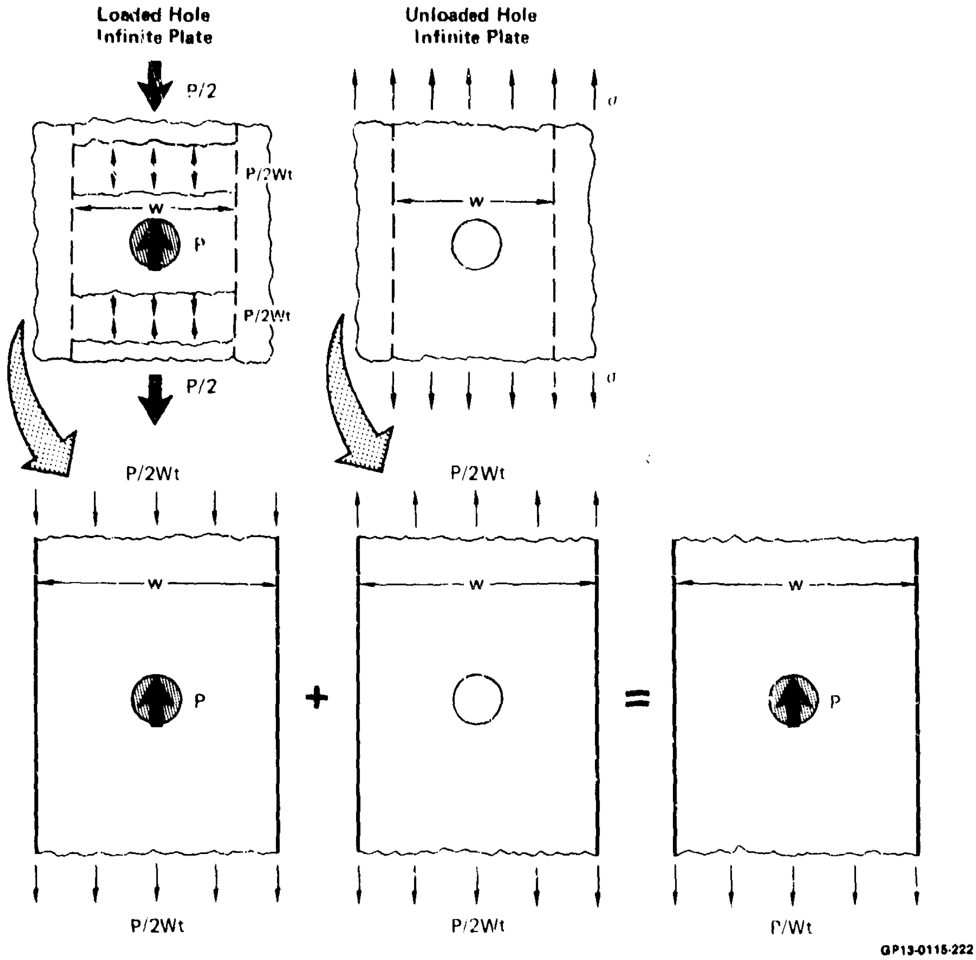

Figure 5. Superposition of Infinite Plate Results Approximates a Finite Width Joint

SECTION III

USER'S INSTRUCTIONS

Conversational interactive procedures are used for specifying input data for operation of the BJSFM program which is programmed to accept free formatted input data. A user may, after becoming familiar with the input procedures, elect to delete input instructions and receive only question marks, identifying all required input data. Using the various output options, users may receive as much or as little data as desired. The nature of required input data is dependent on user-selected output data options; the BJSFM program automatically adjusts its input data requirements to accommodate each output data option.

Up to eight different ply orientations and three different materials may be input. For each ply orientation, a corresponding thickness must be specified as well as material for hybrid laminates. Ply thickness may be either actual or a normalized thickness. A mid-plane symmetric stacking sequence is assumed. Zero degree plies are oriented parallel to the X-axis. Nomenclature applicable to the BJSFM is summarized in Figure 6. Positive angles are measured counterclockwise from the X-axis. All input data units must be consistent.

Unidirectional lamina material stiffness properties are required input data for each different material specified. Unidirectional lamina strength allowables for each material are required only if failure analysis is to be performed. If the maximum strain material failure criterion is used, lamina strain allowables must be input; otherwise, input lamina allowables are in terms of stress.

Any set of in-plane far-field stresses may be applied to an infinite anisotropic or isotropic plate (Figure 1). Bearing stress direction is independent of far-field stress directions.

The BJSFM is only capable of handling finite widths for bolt bearing problems; the width, W, is defined as perpendicular to the bolt load direction (Figure 1). The stresses calculated in a finite-width bolt bearing problem are approximate and most accurate for width-to-diameter ratios greater than four. In combined loading conditions, the finite width routine applies to only the loaded hole portion of the problem. To obtain infinite plate results for a loaded hole, input specimen width as Ø.Ø.

The user must specify the "range" (between low and high) of angular interval between locations around the hole for which data will be calculated. This range must also be subdivided by user-selected "degrees between output" to specify points at which calculations are to be made. "Step increments" are used to obtain data at increasing distances away from the hole boundary.

Coordinate Systems

X - Y	Laminate Axis System Originating at the Center of the Hole. Zero Degree Plies are Parallel to the X-Axis.
X' - Y'	Rotation of the X-Y Axis System for Application of Far-Field Stresses
1 - 2	Lamina Axis System; Fibers are Parallel to the 1 Axis and Transverse to the 2 Axis.

Variable Description (Units)**

E1	Lamina Modulus of Elasticity in Fiber (1) Direction (F/L^2)
E2	Lamina Modulus of Elasticity in Transverse (2) Direction (F/L^2)
G12	Lamina Shear Modulus (F/L^2)
V12	Lamina Poisson's Ratio
EX	Laminate Modulus of Elasticity in X Direction (F/L^2)
EY	Laminate Modulus of Elasticity in Y Direction (F/L^2)
GXY	Laminate Shear Modulus (F/L^2)
VXY	Laminate Poisson's Ratio
T1	Lamina Allowable Tensile Strength in Fiber (1) Direction (F/L^2 or L/L)
C1	Lamina Allowable Compressive Strength in Fiber (1) Direction (F/L^2 or L/L)
T2	Lamina Allowable Tensile Strength in Transverse (2) Direction (F/L^2 or L/L)
C2	Lamina Allowable Compressive Strength in Transverse (2) Direction (F/L^2 or L/L)
S	Lamina Allowable Shear Strength (F/L^2 or L/L)
PX	Stress in X' Direction (F/L^2) - Independent of Input Thickness
PY	Stress in Y' Direction (F/L^2) - Independent of Input Thickness
PXY	Shear Stress (F/L^2) - Independent of Input Thickness
P	Applied Bearing Stress (F/L^2) - P = Bolt Load/(Dia x Actual Thickness)
U	Displacement in X Direction (L)
V	Displacement in Y Direction (L)
W	Specimen Width (L) - Bolt Loading Only
DIST	Radial Distance from Hole Boundary (L)
α	Angle of Applied Bolt Load with X Axis (i.e., Bolt Loading Angle)
β	Rotation Angle of X' - Y' Axes from X - Y Axis System (i.e., Off Axis Loading Angle)
θ	Angle from X Axis to a Point Around Fastener Hole
ϕ	Rotation Angle of 1 - 2 Axes from X - Y Axis System (i.e., Ply Orientation Angle)

All angular measurements are positive counterclockwise from the X axis.

* See also Figure 1.

** Any consistent set of units may be used:

F = Force

L = Length

GP13-0115-223

Figure 6. Program Nomenclature*

The maximum input step increment is seven evenly spaced concentric circles; the first step is always at the hole boundary.

The option to use any one of five different failure criteria has been programmed into the BJSFM. Failure analysis is applied on a ply-by-ply basis; therefore, only unidirectional (lamina) allowables are required input data. Only the maximum strain criterion requires the allowables to be input as strains; all others use stress allowables. Equations for each of the programmed failure criteria are given below. When the right hand side of any of the equations exceeds unity, failure has been predicted for the ply. Tension or compression stress/strain allowables used in each criteria are selected automatically, depending on the sign of individual stress field components being evaluated.

Maximum Strain

$$\frac{\varepsilon_1}{F_1} = 1 \qquad \frac{\varepsilon_2}{F_2} = 1 \qquad \frac{\gamma_{12}}{F_{12}} = 1$$

Maximum Stress

$$\frac{\sigma_1}{F_1} = 1 \qquad \frac{\sigma_2}{F_2} = 1 \qquad \frac{\tau_{12}}{F_{12}} = 1$$

Tsai-Hill

$$\left(\frac{\sigma_1}{F_1}\right)^2 + \left(\frac{\sigma_2}{F_2}\right)^2 + \left(\frac{\tau_{12}}{F_{12}}\right)^2 - \frac{\sigma_1 \sigma_2}{F_1^{\,2}} = 1$$

Modified Tsai-Wu

$$\frac{\sigma_1^{\,2}}{F_1^{\,t} F_1^{\,c}} + \frac{\sigma_2^{\,2}}{F_2^{\,t} F_2^{\,c}} + \left(\frac{1}{F_1^{\,t}} - \frac{1}{F_1^{\,c}}\right)\sigma_1 + \left(\frac{1}{F_2^{\,t}} - \frac{1}{F_2^{\,c}}\right)\sigma_2 + \frac{\tau_{12}^{\,2}}{F_{12}^{\,2}} = 1$$

Hoffman

$$\frac{\sigma_1^{\,2}}{F_1^{\,t}F_1^{\,c}} + \frac{\sigma_2^{\,2}}{F_2^{\,t}F_2^{\,c}} - \frac{\sigma_1\sigma_2}{F_1^{\,t}F_1^{\,c}} + \frac{F_1^{\,c} - F_1^{\,t}}{F_1^{\,t}F_1^{\,c}}\,\sigma_1 + \frac{F_2^{\,c} - F_2^{\,t}}{F_2^{\,t}F_2^{\,c}}\,\sigma_2$$

$$+ \frac{\tau_{12}^{\,2}}{F_{12}^{\,2}} = 1$$

An example printout of the "conversational" language used to request input data is shown in Figure 7.

```
RNH
 1DO YOU WANT INSTRUCTIONS?
?YES
 SELECT DESIRED OUTPUT FROM THE FOLLOWING CASES.
  1  CARPET PLOT DATA
  2  LAMINATE PROPERTIES
  3  LAMINATE STRESSES
  4  LAMINATE STRAINS
  5  CIRCUMFERENTIAL & RADIAL STRESSES/STRAINS
  6  DISPLACEMENTS
  7  STRAINS PER PLY
  8  STRESSES PER PLY
  9  FAILURE CRITERIA PER PLY
 10  AUTOMATIC SEARCH FOR FAILURE
?2,3,4,5,6,7,8,9,10
 INPUT NUMBER OF DIFFERENT PLIES TO BE INPUT (8 MAX) AND
 NUMBER OF DIFFERENT MATERIALS (3 MAX)
?4,1
 INPUT THE UNIDIRECTIONAL MATERIAL PROPERTIES FOR EACH MATERIAL
 IN THE FOLLOWING ORDER:  E1, E2, G12, POISSONS RATIO
?18.85E6,1.9E6,.85E6,.3
 INPUT THE UNIDIRECTIONAL ALLOWABLES FOR EACH MATERIAL
 IN THE FOLLOWING ORDER:  T1, C1, T2, C2, SHEAR
?230000,320000,28200,32300,17300
 INPUT THE ANGULAR ORIENTATION OF EACH PLY
?0.,45.,-45.,90.
 INPUT THE THICKNESS OF EACH PLY
?.5,.2,.2,.1
 INPUT:  FAR FIELD STRESSES PX,PY,PXY, OFF AXIS ANGLE, BEARING STRESS
 AND BOLT LOADING ANGLE.
?10000.,0.,2500.,45.,50000.,10.
 INPUT WIDTH (0.0 FOR INFINITE PLATE)
?1.5
 INPUT BOLT DIAMETER, DEGREES BETWEEN OUTPUT, LOW RANGE, HIGH RANGE,
 STEP INCREMENT AND NUMBER OF STEPS DESIRED (7 MAX)
?.25,30.,0.,360.,.02,2
 INPUT THE NUMBER WHICH CORRESPONDS TO THE FAILURE CRITERIA
 YOU WISH TO USE
 1  MAXIMUM STRAIN
 2  MAXIMUM STRESS
 3  TSAI-HILL
 4  MODIFIED TSAI-WU
 5  HOFFMAN
?3
```

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Figure 7. Example Input Instructions

SECTION IV

OUTPUT OPTIONS

Various output options are available for user selection. The user may select any or all of the following options by inputting the appropriate number(s):

1. Carpet Plot Data
2. Laminate Properties
3. Laminate Stresses
4. Laminate Strains
5. Circumferential and Radial Stresses/Strains
6. Laminate Displacements
7. Strains per Ply
8. Stresses per Ply
9. Failure Criteria per Ply
10. Automatic Search for Failure

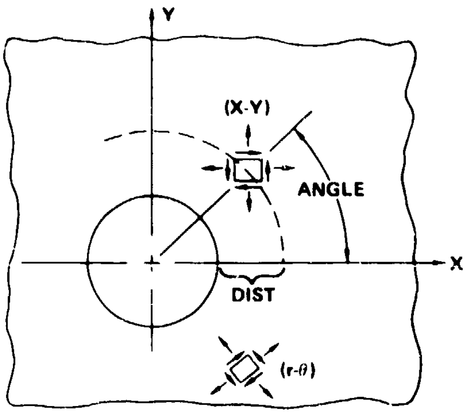
A brief description of each output option follows.

Option #1 - Carpet Plot Data - The carpet plot data routine will automatically vary the layup of a user input $0°/\pm\phi/90°$ laminate family and calculate any one or all of the other output options (2 through 10) for each layup. Sixty-six different layups are automatically calculated in this routine; therefore, large amounts of data will be generated when using this output option.

Option #2 - Laminate Properties - Laminate stiffness properties are calculated using the unidirectional material elastic constants, ply angular orientations and ply thicknesses. These properties are calculated with respect to the X-Y axes and are the same as would be obtained using conventional lamination theory approaches.

Options #3 and #4 - Laminate Stresses/Strains - Laminate stress and strain distributions are available as output at points around the perimeter of the hole and at other user-specified concentric circles about the hole boundary. Principal stresses and strains are also calculated. All output is referenced to the X-Y axes. Points are located by the radial distance away from the hole boundary and the angular orientation from the X-axis (Figure 8).

Option #5 - Circumferential and Radial Stresses/Strains - Circumferential and radial laminate stresses and strains are calculated by a coordinate transformation. Output is in polar coordinates.



EXAMPLE BJSFM OUTPUT:

LAMINATE STRESSES

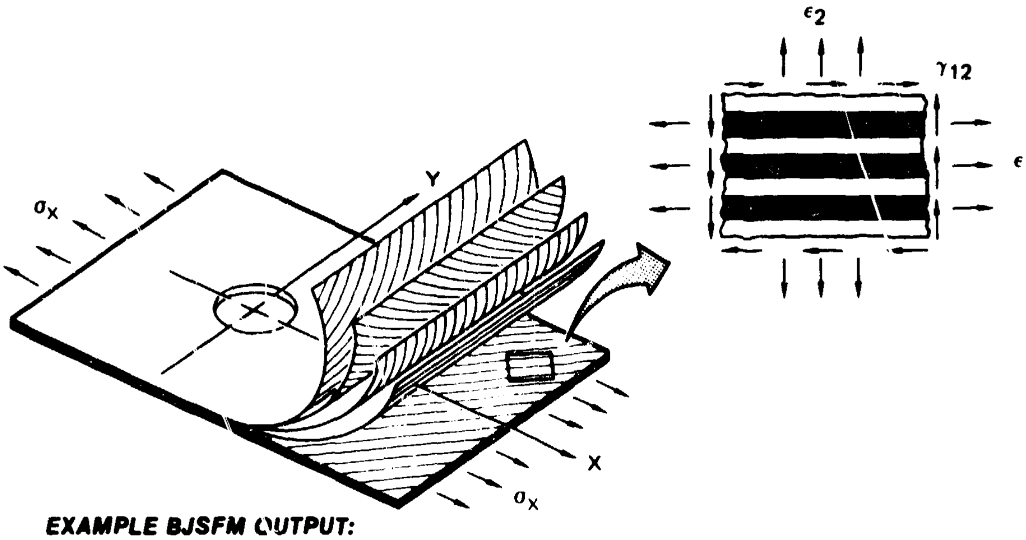
DIST	ANGLE	X STRESS	Y STRESS	SHEAR STRESS	MAX. PRINCIPAL	MIN. PRINCIPAL	DIRECTION
0.000	0.00	-62693.22	44412.49	-9.44	44412.50	-62693.23	.01
0.000	30.00	-40151.39	-813.39	-34070.37	18954.04	-59824.32	[illegible]
0.000	60.00	-6819.06	-29542.53	-19690.77	4552.17	[illegible]	
0.000	90.00	58351.08	-11093.21	-9.44	[illegible]		
0.000	120.00	48122.47	16033.31	27733.47			
[illegible]	150.00	9612.46	29347.34	[illegible]			

GP13-0115-225

Figure 8. Laminate Stress Distribution Output Data Option

<u>Option #6 - Laminate Displacements</u> - Displacements for each point are output as U and V, which are displacements in the X and Y directions respectively. Due to limitations in the derivation, displacements for the loaded hole case shall be considered accurate only within approximately three times the diameter of the fastener. Unloaded hole displacements are exact throughout the plate.

<u>Options #7 and 8 - Strains/Stresses per Ply</u> - Strains and stresses per ply are calculated and output in the lamina (1-2) coordinate system. Each ply is identified along with the location of the point around the hole for which stresses/strains are calculated (Figure 9).



EXAMPLE BJSFM OUTPUT:

STRAINS PER PLY

DIST	ANGLE	PLY	STRAIN 1	STRAIN 2	SHEAR STRAIN
0.000	0.00	0.00	-.007128	.011127	-.000004
0.000	0.00	45.00	.001998	.002001	.018254
0.000	0.00	-45.00	.002001	.001998	-.0182[illegible]
0.000	0.00	90.00	.011127	-.0071[illegible]	[illegible]
0.000	30.00	0.00	-.003509		

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Figure 9. Ply-by-Ply Strain Distribution Output Data Option

<u>Option #9 - Failure Criteria per Ply</u> - Failure criteria per ply option applies the user-selected material failure criterion (max. stress, Tsai-Hill, etc.) to each ply using the input material allowables. The "failure number" obtained as output data indicates the value calculated by the failure criterion using the stress or strain components at a point. A failure number equal to or greater than one predicts ply failure. The program automatically selects tension or compression allowables depending on the sign of individual stress/strain components being evaluated. "Failure ratios" are output which indicate the relative magnitude of contributing stress components to the overall failure number. Therefore, failure can be assessed as to which stress component is most significant. These failure ratios are in terms of the lamina (1-2) coordinate system.

Option #10 - Automatic Search for Failure - The automatic search for failure routine will search over a user-specified range at each angular increment for the most critical single point as calculated by the material failure criterion. Search for failure is only done at the first step increment away from the hole boundary. Therefore, if a search for failure is to be performed at the boundary of the hole, the step increment must be input as Ø.Ø. The program will automatically ratio the input stress field until first ply failure is predicted. Output is the in-plane stresses at which failure is predicted along with the angular orientation of the predicted failure location. Failure numbers are also given for all other plies at the critical ply failure initiation angle. Failure ratios are also output.

SECTION V

PROGRAM LIMITATIONS

The following are the limitations of the BJSFM program.

- o Strictly valid for homogeneous anisotropic flat plates and assumed valid for mid-plane symmetric laminates.
- o Displacements inaccurate at points more than three times the hole diameter away from the hole boundary for loaded hole cases.
- o Stress fields inaccurate for width-to-diameter ratios less than four.
- o Maximum of eight different ply angular orientations (input).
- o Maximum of three different materials for hybrid laminates (input).
- o Maximum of seven steps away from the hole (output).

The following equation must be satisfied to obtain valid output.

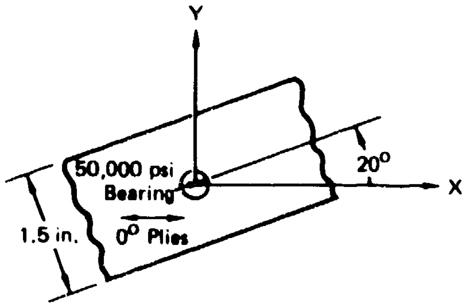
[(High Range)-(Low Range)]/(Degrees Between Output) ≤ 72

The following data must be input as integers:

Output Option Numbers
Number of Different Plies
Number of Different Materials
Material Number
Number of Steps
Failure Criteria Number

SECTION VI

EXAMPLE PROBLEMS




```
1DO YOU WANT INSTRUCTIONS?
?YES
 SELECT DESIRED OUTPUT FROM THE FOLLOWING CASES.
  1  CARPET PLOT DATA
  2  LAMINATE PROPERTIES
  3  LAMINATE STRESSES
  4  LAMINATE STRAINS
  5  CIRCUMFERENTIAL & RADIAL STRESSES/STRAINS
  6  DISPLACEMENTS
  7  STRAINS PER PLY
  8  STRESSES PER PLY
  9  FAILURE CRITERIA PER PLY
 10  AUTOMATIC SEARCH FOR FAILURE
?4,6
 INPUT NUMBER OF DIFFERENT PLIES TO BE INPUT (8 MAX) AND
 NUMBER OF DIFFERENT MATERIALS (3 MAX)
?4,1
 INPUT THE UNIDIRECTIONAL MATERIAL PROPERTIES FOR EAC  MATERIAL
 IN THE FOLLOWING ORDER:  E1, E2, G12, POISSONS RATIO
?19.95E6,1.9E6,.85E6,.3
 INPUT THE ANGULAR ORIENTATION OF EACH PLY
?0.,45.,-45.,90.
 INPUT THE THICKNESS OF EACH PLY          42% - 0° Plies, 50% - ±45°, 8% - 90°
?.42,.25,.25,.08
 INPUT:  FAR FIELD STRESSES PX,PY,PXY, OFF AXIS ANGLE, BEARING STRESS
 AND BOLT LOADING ANGLE.
?0,0,0,0,50000.,20.
 INPUT WIDTH (0.0 FOR INFINITE PLATE)
?1.5
 INPUT BOLT DIAMETER, DEGREES BETWEEN OUTPUT, LOW RANGE, HIGH RANGE,
 STEP INCREMENT AND NUMBER OF STEPS DESIRED (7 MAX)
?.25,30.,0.,90.,.02,3
```

LAMINATE STRAINS (Output Option No. 4)

DIST	ANGLE	X STRAIN	Y STRAIN	SHEAR STRAIN	MAX. PRINCIPAL	MIN. PRINCIPAL	DIRECTION
0.000	0.00	-.006927	.007109	.000001	.007109	-.006927	-.00
0.000	30.00	-.004137	.001938	-.012340	.005705	-.008005	38.20
0.000	60.00	.001704	-.006583	-.010110	.004099	-.008979	-33.87
0.000	90.00	.005512	-.006573	.000001	.005512	-.006573	.01
.020	0.00	-.005642	.005766	.000253	.005767	-.005643	-1.27
.020	30.00	-.003419	.001412	-.009993	.004546	-.006554	38.21
.020	60.00	.001920	-.005302	-.008304	.003762	-.007443	-32.82
.020	90.00	.003843	-.004336	.001880	.003949	-.004442	12.34
.040	0.00	-.004698	.004788	.000437	.004793	-.004703	-2.63
.040	30.00	-.002855	.001130	-.008420	.003795	-.005520	38.34
.040	60.00	.001898	-.004792	-.006542	.003231	-.005125	-31.46
.040	90.00	.002845	-.003118	.001857	.002986	-.003259	15.96

DISPLACEMENTS (Output Option No. 6)

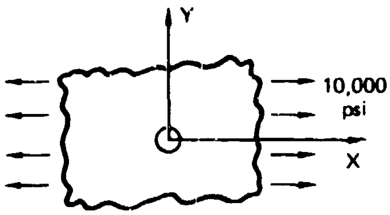
DIST	ANGLE	U	V
0.000	0.00	.000794	-.000080
0.000	30.00	.000943	.000342
0.000	60.00	.000691	.000535
0.000	90.00	.000215	.000294
.020	0.00	.000655	-.000093
.020	30.00	.000794	.000300
.020	60.00	.000518	.000471
.020	90.00	.000058	.000234
.040	0.00	.000538	-.000104
.040	30.00	.000668	.000268
.040	60.00	.000378	.000419
.040	90.00	-.000045	.000192

```
 DO YOU WISH TO CONTINUE?
?NO
   STOP
```

GP13-0116-220

Figure 10. Loaded Hole Case

```
1DO YOU WANT INSTRUCTIONS?
?YES
 SELECT DESIRED OUTPUT FROM THE FOLLOWING CASES.
  1  CARPET PLOT DATA
  2  LAMINATE PROPERTIES
  3  LAMINATE STRESSES
  4  LAMINATE STRAINS
  5  CIRCUMFERENTIAL & RADIAL STRESSES/STRAINS
  6  DISPLACEMENTS
  7  STRAINS PER PLY
  8  STRESSES PER PLY
  9  FAILURE CRITERIA PER PLY
 10  AUTOMATIC SEARCH FOR FAILURE
?3,5
 INPUT NUMBER OF DIFFERENT PLIES TO BE INPUT (8 MAX) AND
 NUMBER OF DIFFERENT MATERIALS (3 MAX)
?1,1
 INPUT THE UNIDIRECTIONAL MATERIAL PROPERTIES FOR EACH MATERIAL
 IN THE FOLLOWING ORDER:  E1, E2, G12, POISSONS RATIO
?10.E6,10.E6,3.85E6,.3
 INPUT THE ANGULAR ORIENTATION OF EACH PLY
?0.
 INPUT THE THICKNESS OF EACH PLY
?1.
 INPUT:  FAR FIELD STRESSES PX,PY,PXY, OFF AXIS ANGLE, BEARING STRESS
 AND BOLT LOADING ANGLE.
?10000.,0.,0.,0.,0.,0.
 INPUT BOLT DIAMETER, DEGREES BETWEEN OUTPUT, LOW RANGE, HIGH RANGE,
 STEP INCREMENT AND NUMBER OF STEPS DESIRED (7 MAX)
?.25,15.,0.,90.,0.,1
```



Isotropic Properties

$$\left(E1 = E2,\ G = \frac{E1}{2\,(1+\nu)}\right)$$

LAMINATE STRESSES (Output Option No. 3)

DIST	ANGLE	X STRESS	Y STRESS	SHEAR STRESS	MAX. PRINCIPAL	MIN. PRINCIPAL	DIRECTION
0.000	0.00	-.00	-10000.00	0.00	-.00	-10000.00	0.00
0.000	15.00	-490.49	-6831.64	1[illegible]0.53	-.00	-7322.13	15.00
0.000	30.00	-.41	-1.22	.70	-.00	-1.62	30.00
0.000	45.00	5001.62	5001.62	-5001.62	10003.25	-.00	45.00
0.000	60.00	15003.65	5001.22	-86[illegible]2.36	20004.87	.00	-30.00
0.000	75.00	25480.87	1830.02	-68[illegible]9.72	27318.88	.00	-15.00
0.000	90.00	29993.51	-.00	-.00	29993.51	-.00	-.00

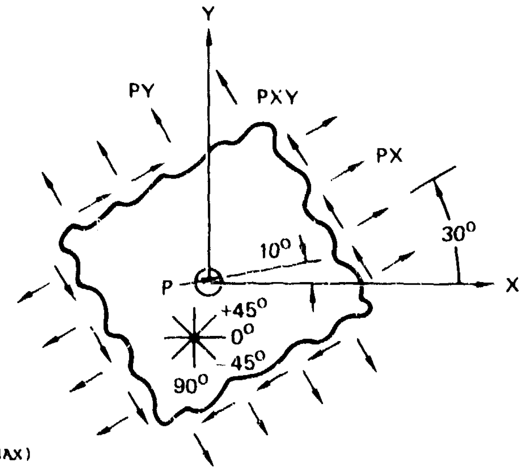
CIRCUMFERENTIAL AND RADIAL STRESSES & STRAINS (Output Option No. 5)

DIST	ANGLE	THETA STRESS	RADIAL STRESS	SHEAR STRESS	THETA STRAIN	RADIAL STRAIN	SHEAR STRAIN
0.000	0.00	-10000.00	-.00	0.00	-.001000	.000300	0.000000
0.000	15.00	-7322.13	-.00	-.00	-.000732	.000220	.000206
0.000	30.00	-1.62	-.00	.00	-.000000	.000000	.000000
0.000	45.00	10003.25	-.00	.00	.001000	-.000299	-.000000
0.000	60.00	20004.87	.00	.00	.002000	-.000599	.000562
0.000	75.00	27318.88	.00	.00	.002731	-.000819	.000767
0.000	90.00	29993.51	-.00	-.00	.002999	-.000900	.000000

```
 DO YOU WISH TO CONTINUE?
?NO
    STOP
```

GP13-0115-221

Figure 11. Isotropic Unloaded Hole




```
1DO YOU WANT INSTRUCTIONS?
?YES
 SELECT DESIRED OUTPUT FROM THE FOLLOWING CASES.
  1   CARPET PLOT DATA
  2   LAMINATE PROPERTIES
  3   LAMINATE STRESSES
  4   LAMINATE STRAINS
  5   CIRCUMFERENTIAL & RADIAL STRESSES/STRAINS
  6   DISPLACEMENTS
  7   STRAINS PER PLY
  8   STRESSES PER PLY
  9   FAILURE CRITERIA PER PLY
 10   AUTOMATIC SEARCH FOR FAILURE
?10
 INPUT NUMBER OF DIFFERENT PLIES TO BE INPUT (8 MAX)
 AND NUMBER OF DIFFERENT MATERIALS (3 MAX)
?4,1
 INPUT THE UNIDIRECTIONAL MATERIAL PROPERTIES FOR EACH
 MATERIAL IN THE FOLLOWING ORDER:  E1, E2, G12, POISSONS RATIO
?18.85E6,1.9E6,.85E6,.3
 INPUT THE UNIDIRECTIONAL ALLOWABLES FOR EACH MATERIAL
 IN THE FOLLOWING ORDER:  T1, C1, T2, C2, SHEAR
?2.3E5,3.2E5,2.82E4, .23E4,1.73E4
 INPUT THE ANGULAR ORIENTATION OF EACH PLY
?0.,45.,-45.,90.
 INPUT THE THICKNESS OF EACH PLY
?.5,.2,.2,.1
 INPUT:  FAR FIELD STRESSES PX,PY,PXY, OFF AXIS ANGLE, BEARING STRESS
 AND BOLT LOADING ANGLE.
?10000.,500.,2500.,30.,35000.,10.
 INPUT WIDTH (0.0 FOR INFINITE PLATE)
?0.0
 INPUT BOLT DIAMETER, DEGREES BETWEEN OUTPUT, LOW RANGE, HIGH RANGE,
 STEP INCREMENT AND NUMBER OF STEPS DESIRED (7 MAX)
?.25,5.,0.,360.,.02,2
 INPUT THE NUMBER WHICH CORRESPONDS TO THE FAILURE CRITERIA
 YOU WISH TO USE
 1   MAXIMUM STRAIN
 2   MAXIMUM STRESS
 3   TSAI-HILL
 4   MODIFIED TSAI-WU
 5   HOFFMAN
?3
```

Input Stresses are Ratioed Uniformly Until First Ply Failure is Predicted

AUTOMATIC SEARCH FOR FAILURE (Output Option No. 10)

FAILURE STRESSES

PX	PY	PXY	P
17243.40	862.17	4310.85	60351.88

DIST	ANGLE	PLY	FAILURE NUMBER	FAILURE RATIOS	2	SHEAR
.020	335.00	0.00	.355	-.210	.469	.353
.020	335.00	45.00	.999	.811	-.181	.537
.020	335.00	-45.00	.730	-.400	.650	-.628
.020	335.00	90.00	1.000	.606	-.075	-.788

Indicates Ply Failure

Depicts Relative Magnitude of Stress Components to Failure Number

Failure Initiation Location Angle

User Specified Distance Away from Hole at Which Failure is Predicted

Note: The "search" is only done within the user specified "range" with an accuracy of 1/2 the "degrees between output" in locating failure initiation.

GP13-0115-219

Figure 12. General Loading Condition