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**Abstract**—The abstract goes here.

**Keywords**—*Keyword1; Keyword2; Keyword3*

## I. INTRODUCTION

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### A. Subsection Heading Here

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## II. INTRODUCTION

Composite materials offer improved strength, stiffness, corrosion resistance, etc. over conventional materials, and are widely used as alternative materials for applications in various industries ranging from electronic packaging to golf clubs, and medical equipment to homebuilding, making aircraft structure to space vehicles. One widely known advantage of using composite material is can significantly reducing the weight of target structure.

thickness[1], [2], [3], [4], [5], [6], [7], [8], [9], [10], [11], [12], [13], [14], [15], [16], [17], [18], [19], [20], [21]

Genetic algorithm(GA)[22], [18], [23], [19], [24], [25], [26], [27], [28], [?]

Colony Optimization[29], [?]

In practice, fiber orientations are restricted to a finite set of angles, and ply thickness is a specific numeric value. Because the design variables are not continuous, a gradient based optimization procedure, such as gradient descent method, is not suitable to cope with such problems. Moreover, gradient based optimization approach is very easily to get trapped in local minima, and many local optimum may exist in structural optimization problems. A stochastic optimization, such as genetic algorithm(GA) and simulated annealing(SA), is able to deal with optimization problem with discrete variables. Besides, stochastic method could escape from local optimum, and obtain global optimum. GA is one of the most reliably stochastic algorithm, which has been widely used in solving

constraint design for composite laminate. Although GA gains different advantages for solving discrete problems, many disadvantages exist within this approach. First, the optimization process of GA parameters, such as the population size, parent population, mutation percentage, etc., is very tedious; Second, the GA needs to evaluate the objective functions many times to achieve the optimization, and the computation cost is very high; the last problem within GA is the premature convergence. GA consists of five basic parts: the variable coding, selection scheme, crossover operator, mutation operator and how the constraints are handled.

GA is originally proposed for unconstrained optimization. However, in order to deal with constrained design for composite laminate, some techniques were introduced into the GA. The first method is using of data structure, special data structure was developed to fulfill the symmetry constraint of the laminate, which consists of coding only half of the laminate and considering that each stack of the laminate is formed by two laminae with the same orientation but opposite signs[5], [30]. A penalty function is developed to convert a constrained problem into an unconstrained problem by adding penalty term to the objective function. Another method to solve constraint problem is introducing repair strategy by Todoroki and Haftka [31], which is aimed to transform infeasible solutions to feasible solution.

The first issue when implementing a GA is

To check the feasibility of a laminate composite by imposing a strength constraint, various failure criteria have been proposed to decide whether it fails or not. Rankine and Tresca invented maximum normal stress theory and maximum shearing stress theory, respectively, also known as Maximum stress theory. St. Venant and Tresca proposed maximum normal strain theory and maximum shear theory, respectively, also known as Maximum strain failure theory. Tsai-Hill Failure theory is came up based on the distortion energy failure theory of Von-Mises's distortional energy yield criterion for isotropic materials.

The rest of the paper is organized as follows. Section 2 explains the classical laminate theory and the failure criteria taken in the present study. Section 3 explains the proposed method of selection strategy and self-adaptive parameters for mutation during the GA process. Section 4 describes the result of the numerical experiments in different cases, and in the Conclusion section we discuss the results.

### III. ANALYSIS OF

#### A. Stress and Strain in a Lamina

For a single lamina has a small thickness under plane stress, and it's upper and lower surfaces of the lamina are free from external loads. According to the Hooke's Law, the three-dimensional stress-strain equations can be reduced to two-dimensional stress-strain equations. The stress-strain relation in local axis 1-2 is:

$$\begin{bmatrix} \sigma_1 \\ \sigma_2 \\ \tau_{12} \end{bmatrix} = \begin{bmatrix} Q_{11} & Q_{12} & 0 \\ Q_{12} & Q_{22} & 0 \\ 0 & 0 & Q_{66} \end{bmatrix} \begin{bmatrix} \varepsilon_1 \\ \varepsilon_2 \\ \gamma_{12} \end{bmatrix} \quad (1)$$

where  $Q_{ij}$  are the stiffnesses of the lamina that are related

to engineering elastic constants given by

$$\begin{aligned} Q_{11} &= \frac{E_1}{1 - \nu_{12}\nu_{21}} \\ Q_{22} &= \frac{E_2}{1 - \nu_{12}\nu_{21}} \\ Q_{66} &= G_{12} \\ Q_{12} &= \frac{\nu_{21}E_2}{1 - \nu_{12}\nu_{21}} \end{aligned} \quad (2)$$

where  $E_1, E_2, \nu_{12}, G_{12}$  are four independent engineering elastic constants, which are defined as follows:  $E_1$  is the longitudinal Young's modulus,  $E_2$  is the transverse Young's modulus,  $\nu_{12}$  is the major Poisson's ratio, and  $G_{12}$  is the in-plane shear modulus.

Stress strain relation in the global x-y axis:

$$\begin{bmatrix} \sigma_x \\ \sigma_y \\ \tau_{xy} \end{bmatrix} = \begin{bmatrix} \bar{Q}_{11} & \bar{Q}_{12} & \bar{Q}_{16} \\ \bar{Q}_{12} & \bar{Q}_{22} & \bar{Q}_{26} \\ \bar{Q}_{16} & \bar{Q}_{26} & \bar{Q}_{66} \end{bmatrix} \begin{bmatrix} \varepsilon_x \\ \varepsilon_y \\ \gamma_{xy} \end{bmatrix} \quad (3)$$

where

$$\begin{aligned} \bar{Q}_{11} &= Q_{11}c^4 + Q_{22}s^4 + 2(Q_{12} + 2Q_{66})s^2c^2 \\ \bar{Q}_{12} &= (Q_{11} + Q_{22} - 4Q_{66})s^2c^2 + Q_{12}(c^4 + s^4) \\ \bar{Q}_{22} &= Q_{11}s^4 + Q_{22}c^4 + 2(Q_{12} + 2Q_{66})s^2c^2 \\ \bar{Q}_{16} &= (Q_{11} - Q_{12} - 2Q_{66})c^3s - (Q_{22} - Q_{12} - 2Q_{66})s^3c \\ \bar{Q}_{26} &= (Q_{11} - Q_{12} - 2Q_{66})cs^3 - (Q_{22} - Q_{12} - 2Q_{66})c^3s \\ \bar{Q}_{66} &= (Q_{11} + Q_{22} - 2Q_{12} - 2Q_{66})s^2c^2 + Q_{66}(s^4 + c^4) \end{aligned} \quad (4)$$

The c and s denotes  $\cos\theta$  and  $\sin\theta$ .

The local and global stresses in an angle lamina are related to each other through the angle of the lamina  $\theta$

$$\begin{bmatrix} \sigma_1 \\ \sigma_2 \\ \tau_{12} \end{bmatrix} = [T] \begin{bmatrix} \sigma_x \\ \sigma_y \\ \tau_{xy} \end{bmatrix} \quad (5)$$

where

$$[T] = \begin{bmatrix} c^2 & s^2 & 2sc \\ s^2 & c^2 & -2sc \\ -sc & sc & c^2 - s^2 \end{bmatrix} \quad (6)$$

#### B. Stress and Strain in a Laminate

$$\begin{aligned} \begin{bmatrix} N_x \\ N_y \\ N_{xy} \end{bmatrix} &= \begin{bmatrix} A_{11} & A_{12} & A_{16} \\ A_{12} & A_{22} & A_{26} \\ A_{16} & A_{26} & A_{66} \end{bmatrix} \begin{bmatrix} \varepsilon_x^0 \\ \varepsilon_y^0 \\ \gamma_{xy}^0 \end{bmatrix} \\ &+ \begin{bmatrix} B_{11} & B_{12} & B_{16} \\ B_{11} & B_{12} & B_{16} \\ B_{16} & B_{26} & B_{66} \end{bmatrix} \begin{bmatrix} k_x \\ k_y \\ k_{xy} \end{bmatrix} \\ \begin{bmatrix} M_x \\ M_y \\ M_{xy} \end{bmatrix} &= \begin{bmatrix} B_{11} & B_{12} & B_{16} \\ B_{12} & B_{22} & B_{26} \\ B_{16} & B_{26} & B_{66} \end{bmatrix} \begin{bmatrix} \varepsilon_x^0 \\ \varepsilon_y^0 \\ \gamma_{xy}^0 \end{bmatrix} \\ &+ \begin{bmatrix} D_{11} & D_{12} & D_{16} \\ D_{11} & D_{12} & D_{16} \\ D_{16} & D_{26} & D_{66} \end{bmatrix} \begin{bmatrix} k_x \\ k_y \\ k_{xy} \end{bmatrix} \end{aligned} \quad (7)$$

$N_x, N_y$  - normal force per unit length

$N_{xy}$  - shear force per unit length

$M_x, M_y$  - bending moment per unit length

$M_{xy}$  - twisting moments per unit length

$\varepsilon^0, k$ - mid plane strains and curvature of a laminate in x-y coordinates

The mid plane strain and curvature is given by

$$\begin{aligned} A_{ij} &= \sum_{k=1}^n (\bar{Q}_{ij})_k (h_k - h_{k-1}) i = 1, 2, 6, j = 1, 2, 6 \\ B_{ij} &= \frac{1}{2} \sum_{k=1}^n (\bar{Q}_{ij})_k (h_k^2 - h_{k-1}^2) i = 1, 2, 6, j = 1, 2, 6 \\ D_{ij} &= \frac{1}{3} \sum_{k=1}^n (\bar{Q}_{ij})_k (h_k^3 - h_{k-1}^3) i = 1, 2, 6, j = 1, 2, 6 \end{aligned} \quad (8)$$

The [A], [B], and [D] matrices are called the extensional, coupling, and bending stiffness matrices, respectively. The extensional stiffness matrix [A] relates the resultant in-plane forces to the in-plane strains, and the bending stiffness matrix [D] couples the resultant bending moments to the plane curvatures. The coupling stiffness matrix [B] relates the force and moment terms to the midplane strains and midplane curvatures.

### IV. FAILURE CRITERIA FOR A LAMINA

Many different criteria about the failure of an angle lamina have been proposed for a unidirectional lamina, such as the maximum stress failure theory, maximum strain failure theory, Tsai-Hill failure theory, and Tsai-Wu failure theory. The failure criterion of a lamina are based on the stresses in the local axes instead of principal normal stresses and maximum shear stresses. There are four normal strength parameters and one shear stress for a unidirectional lamina. The five strength parameters are

$(\sigma_1^T)_{ult}$  = ultimate longitudinal tensile strength(in direction 1),

$(\sigma_1^C)_{ult}$  = ultimate longitudinal compressive strength(in direction 1),

$(\sigma_2^T)_{ult}$  = ultimate transverse tensile strength(in direction 2),

$(\sigma_2^C)_{ult}$  = ultimate transverse compressive strength(in direction 2), and

$(\tau_{12})_{ult}$  = and ultimate in-plane shear strength(in plane 12).

In the present study, Tsai-wu failure theory is taken to decide whether a lamina fails, because this theory is more general than the Tsai-Hill failure theory, which considers two different situations, the compression and tensile strengths of a lamina. A lamina is considered to fail if

$$H_1\sigma_1 + H_2\sigma_2 + H_6\tau_{12} + H_{11}\sigma_1^2 + H_{22}\sigma_2^2 + H_{66}\tau_{12}^2 + 2H_{12}\sigma_1\sigma_2 < 1 \quad (9)$$

is violated, where

$$SR = \frac{\text{Maximum Load Which Can Be Applied}}{\text{Load Applied}} \quad (10)$$

Maximum stress failure theory consists of maximum normal stress theory proposed by Rankine and maximum shearing stress theory by Tresca. The stresses applied on a lamina can be resolved into the normal and shear stresses in the local axes. If any of the normal or shear stresses in the local axes of a lamina is equal or exceeds the corresponding ultimate strengths of the unidirectional lamina, the lamina is considered to be failed.

#### A. Failure Theories for a Laminate

If keep increasing the loading applied to a laminate, the laminate will fails. The failure process of a laminate is more complicate than a lamina, because a laminate consists of multiple plies, and the fiber orientation, material, thickness of each ply maybe different from the others. In most situations, some layer fails first and the remains continue to take more loads until all the plies fail. If one ply fails, it means this lamina does not contribute to the load carrying capacity of the laminate. The procedure for finding the first failure ply given follows the fully discounted method:

- 1) Compute the reduced stiffness matrix  $[Q]$  referred to as the local axis for each ply using its four engineering elastic constants  $E_1$ ,  $E_2$ ,  $E_{12}$ , and  $G_{12}$ .
- 2) Calculate the transformed reduced stiffness  $[\bar{Q}]$  referring to the global coordinate system (x, y) using the reduced stiffness matrix  $[Q]$  obtained in step 1 and the ply angle for each layer.
- 3) Given the thickness and location of each layer, the three laminate stiffness matrices  $[A]$ ,  $[B]$ , and  $[D]$  are determined.
- 4) Apply the forces and moments,  $[N]_{xy}$ ,  $[M]_{xy}$  solve Equation 7, and calculate the middle plane strain  $[\sigma^0]_{xy}$  and curvature  $[k]_{xy}$ .
- 5) Determine the local strain and stress of each layer under the applied load.
- 6) Use the ply-by-ply stresses and strains in the Tsai-wu failure theory to find the strength ratio, and the layer with smallest strenght ratio is the first failed ply.

## V. METHODOLOGY

### A. Objective function

There are two design variables here, the angles in the laminate, and the number of layers that each fiber orientation has. The objective function is as

$$F = 2t_0 \sum_{k=1}^n n_k$$

The first term represent the total thickness of the composite laminates,  $t_0$  is ply thickness;  $n_k$  is the number of plies in the kth lamina, in which the fiber orientation is  $\theta_k$ .

The only constraint is the safety factor of the material under certain loading, and it should greater than 1.

### B. Selection

The purpose of the selection operator is how to chose parents to produce children of better fitness. Traditional methods of selecting strategies only take the fitness of the individual into account, however, because of the existance of constraint, the selection strategies have to change a little bit. The parents of next generation consists of three groups: proper groups, active groups, and potential groups.

Proper parents mean individual fullfils the constraint, which are chosen by the individual's fitnees, individuals with better fitness are more likely to be chosen if they fit the constraint; active groups means that individual is supposed to be always exist in the parents during the GA, which are selected by fitness, ignoring the constraint; potential groups means that they are likely to turn into proper individual after a couple of generations, and potential individuals are chosen by constraint function, the more the individual fulfils the constraint, the more possiblity it will be selected.

### C. Crossover

The crossover operator happens among these three groups. the child of two proper groups are more likely to be a proper individual which can be used to obtain a better individual. the child of an active individual and a potential individual can significantly change the gene of active individual's chromosome, which lets the individual evolved toward a new direction. The offspring of two active individuals are more likely to be an active individual, which can maintain the active group.

### D. Mutation

A mutation direction is imposed on the mutation operator which to make sure the individual evolving toward the right direction. The mutation direction, denoted by  $md$ , is a n dimensional vector corresponsing to number of constraints, it is decided by the constraint thresholds  $CT_i$  and the current individual's constraint value, denoted as  $CV_i$ . The mutation vector can be obtained by the following formula

$$md = [CT_1, \dots, CT_{n-1}, CT_n] - [CV_0, \dots, CV_{n-1}, CV_n]$$

During the operator, the mutation consists of three parts, the length of the chromosome, the angle of the chromosome, and the number of each angle. Becasue the chromosome's length is positive correlated with the individual's fitness, the coefficient of length mutation denoted by  $C_l$ , if  $\sum_{i=1}^N CT_i$

great than zero, the mutation length is restricted to the range  $[0, C_l \sum_{i=1}^N CT_i]$ ; if the  $\sum_{i=1}^N CT_i$  less than zero, the mutation length is restricted to the range  $[0, \sum_{i=1}^N CT_i]$ ; Assuming a  $[13_6 / -27_4]_s$  carbon T300/5308 composite laminate under the loading  $N_{xx} = N_{yy} = 10$  MPa m, the only constraint is the safety factor greater than 1. According to the Tsai-Wu criterion, its safety factor is 0.0539. So the mutation vector is  $[0.941]$ , assuming the coefficient is 20, so the mutation range is from 0 to 18. A random number is generated from the range  $[0, 18]$ , supposing the outcome is 13, then a length generator is used to a list, the it's sum is 13, suppose the list is  $[5, 8]$ , the laminate after mutation is  $[13_{11} / -27_{12}]_s$ .

The relationship between the angles in the composite laminate and the chromosome's fitness is unclear, so the mutation direction of chromosome's angle is random. The coefficient angle mutation is  $C_a$ ,  $[0, C_a \sum_{i=1}^N CT_i]$

## VI. RESULT

In this experiment, two constraints are imposed on the composite laminates which are the safety factor  $CT_1$  for Tsai-Wu, and safety factor  $CT_2$  for maximum stress. Both of them value is 1. The constraint values of an individual are  $CV_1$  and  $CV_2$ . So the mutation vector here is a two dimensional vector  $[1 - CV_1, 1 - CV_2]$ , and the coefficient of length mutation  $C_l$  and angle mutation  $C_a$ , respectively, chosen here is 20 and 10.

Figure 1 (a) shows how the optimal individual's fitness and strength ratio vary during the GA process. The method to chose optimal individual considering two following situations, if no individual in the current population meets constraint, the one with biggest fitness is selected as the optimal individual; if there are one or multiple individuals fullfils requirement, the one with smallest fitness is chosen. Figure 1 (b) shows how the two distinct fiber orientation changes at the same time, and Figure 1 (c) how the number of each angles change.

At the beginning of this GA process, the fitness curves increased very quickly, because of individual's strength ratio  $CT_0$  is very small, so the difference between the individual's fitness and the imposed constraint threshold is a big positive number, so the range of mutaion length is from 0 to  $C_l(CT_0 - CV_0)$ . The length of individual increases by n, which is random number between 0 and  $C_l(CT_0 - CV_0)$ . As can be seen from Figure 1 (a), both of optimal individual's fitness and strength ratio increases very quickly. The range of mutaion angle is from 0 to  $C_a(CT_0 - CV_0)$ , and the number of every angle also change violently. During this stage, increasing individual's length playing a major role in increasing individual's fitness.

After a couple of generations, the optimal individual's fitness get bigger, and the difference between individual's fitness and constraint threshold get smaller. The range of mutaion length  $[0, C_l(CT_0 - CV_0)]$  turn smaller. At this stage, simply increase the individual's length doesn't make much difference in improve individual's fitness, and a better composite laminates lay-up can dramatically change the optimal individual's fitness. That's why the fitness curve oscillated violently in this stage. At the same time, the strength ratio curve kept growing smoothly. But the growing speed got more smaller.

When GA comes to its last phase, GA found individuals that meet the constraint. Now the optimal individual's fitness

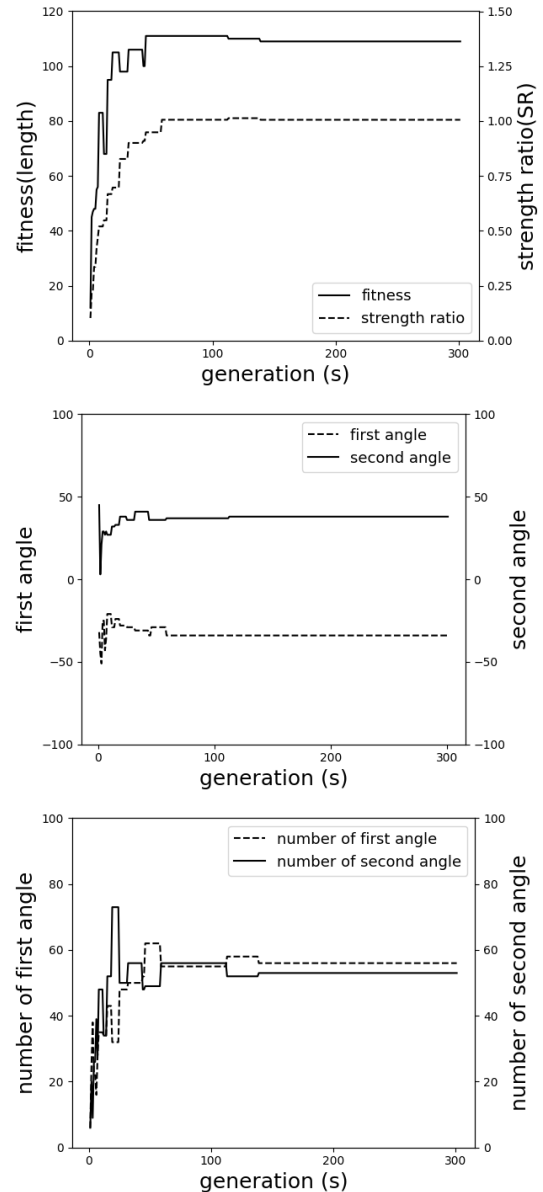


Fig. 1: Two distinct angles

is greater than the safety factor. The range of mutation length is from  $C_l(CT_0 - CV_0)$  to 0. It means individuals need to decrease it's length and improve its internal structure to meet the constraint. That's why the fitness of optimal individual kept decreasing, however, the strength ratio curve still is greater then safety factor.

## VII. CONCLUSION

In this paper, SAGA is proposed to search the optimal lay-up for laminated composite under different loading. Two situations are considered under the same loading, a set of two distinct angles, and three distinct angles, SAGA can adjust its convergence speed depending on the difference of individual's constraint value and constraint threshold.

TABLE I: An Example of a Table

Property	Symbol	Unit	Graphite/Epoxy
Longitudinal elastic modulus	$E_1$	GPa	181
Traverse elastic modulus	$E_2$	GPa	10.3
Major Poisson's ratio	$\nu_{12}$		0.28
Shear modulus	$G_{12}$	GPa	7.17
Ultimate longitudinal tensile strength	$(\sigma_1^T)_{ult}$	MP	1500
Ultimate longitudinal compressive strength	$(\sigma_1^C)_{ult}$	MP	1500
Ultimate transverse tensile strength	$(\sigma_2^T)_{ult}$	MPa	40
Ultimate transverse compressive strength	$(\sigma_2^C)_{ult}$	MPa	246
Ultimate in-plane shear strength	$(\tau_{12})_{ult}$	MPa	68
Density	$\rho$	$g/cm^3$	1.590

TABLE II: The optimum lay-ups using two distinct fiber angles under various biaxial loading cases

Loading $N_x/N_y/N_{xy}$ (MPa m)	Optimum lay-up sequences	laminate thickness	Safety factor	MS
10/5/0	$[33_{29}/-39_{25}/-39]_s$	109	1.0074	1.0246
20/5/0	$[33_{22}/-31_{24}]_s$	92	1.0055	1.2065
40/5/0	$[29_{18}/-21_{23}/-21]_s$	83	1.0034	1.7350
80/5/0	$[-20_{27}/21_{25}/25]_s$	105	1.0029	1.2063
120/5/0	$[-18_{34}/17_{36}]_s$	140	1.0000	1.0898

TABLE III: The optimum lay-ups using three distinct fiber angles under various biaxial loading cases

Loading $N_x/N_y/N_{xy}$ (MPa m)	Optimum lay-up sequences	laminate thickness	Safety factor
10/5/0	$[37_{27}/-38_{27}/-5]_s$	110	1.0023
20/5/0	$[34_{24}/-32_{14}/-28_{11}]_s$	98	1.0237
40/5/0	$[21_{28}/-32_{19}/23]_s$	100	1.0788
80/5/0	$[-21_{25}/-16_3/21_{26}]_s$	108	1.0128
120/5/0	$[-18_{34}/17_{36}]_s$	140	1.0000

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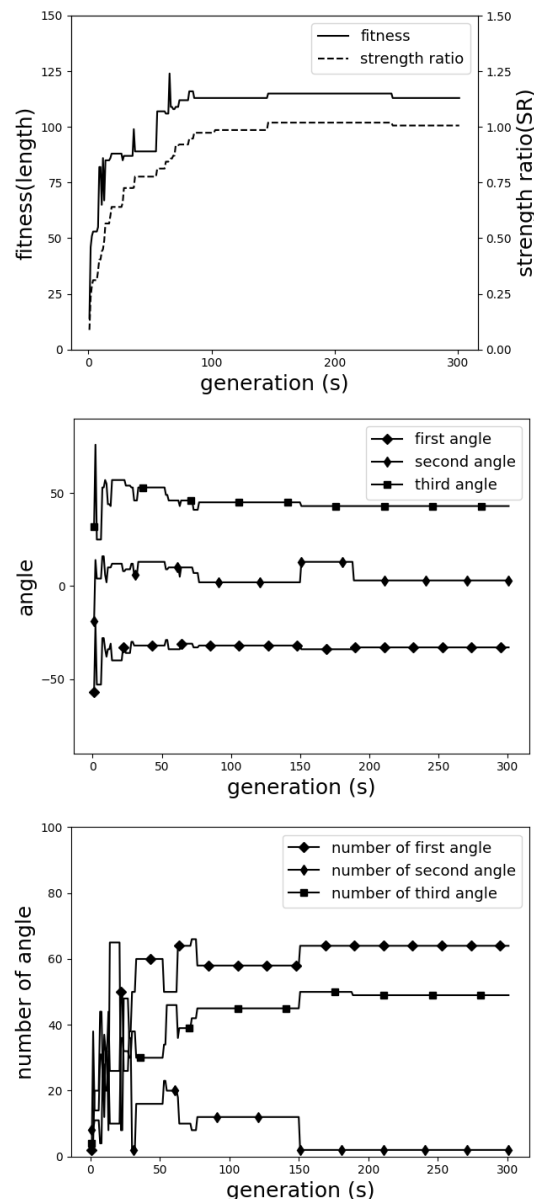


Fig. 2: Three distinct angles

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