

# QUALIFICATION METHODOLOGY FOR ISO26262 CERTIFICATION OF AUTOMOTIVE SoC SYSTEMS

by

TIZIANO FIORUCCI

(Under the Direction of Giorgio di Natale)

## ABSTRACT

This thesis proposes to set up a flow and a methodology of ISO26262 certification for system-type integrated circuits on a digital chip dedicated to driving. These circuits are generally composed of several Intellectual Properties, IPs, dedicated to different functions such as communication or processing of information from sensors (camera, lidar ...), real-time system, vision and imaging, system management (operating system), security. The ISO26262 methodology requires the extraction of a number of metrics related to the resilience of the system to single and multiple faults as well as the effectiveness of countermeasures (detection, reporting and correction of errors) and failure modes. The extraction of failure metrics from fault trees is a method known and documented in the literature. Nevertheless, its application has often been limited to macroscopic electromechanical systems such as a car, actuator or sensor chains. On the other hand, these methods are rarely applied in the field of automotive SoCs where the extraction of metrics is still largely manual (usually using a spreadsheet) and dependent on an expert, and where the verification of the effectiveness of countermeasures is best done by targeted fault injection on a few sub-parts of the complete system or irradiation under a particle beam. This thesis proposes to develop a reliability metrics extraction methodology based on fault injection per block as well as composition methods to obtain the metrics at the level of the complete system. The first part of the thesis will be devoted to the study of the bibliography on the construction of fault trees, the ISO26262 standard and the declination of the different reliability metrics in the case of a digital SoCs type system. The extraction of metrics at the block level will be based on 2 different methods, one analytical based on probabilities, the other experimental based on fault injection. The aim is not to develop new probability codes or fault injection tools

but to develop a methodology to use them in the context of a SoC to obtain the desired data. The second part of the thesis will concern the composition of the data obtained at the functional block level in order to obtain the ISO26262 metrics at the system level (SoC). It will be a matter of developing a composition method adapted in particular to the characteristics of SoCs (communicating system, performing calculations that must react in real time, ...) and to the fault models that characterize them or imposed by the ISO26262 standard. The third part of the thesis concerns the application of the developments described in the previous paragraph to an SoC-type system and the verification of the results obtained.

**French Translation** Cette thèse propose de mettre en place un flux et une méthodologie de certification ISO26262 pour les circuits intégrés de type système sur une puce numérique dédiée à la conduite. Ces circuits sont généralement composés de plusieurs propriétés intellectuelles, IP, dédiées à différentes fonctions telles que la communication ou le traitement d'informations provenant de capteurs (caméra, lidar...), le système en temps réel, la vision et l'imagerie, la gestion du système (système d'exploitation), la sécurité. La méthodologie ISO26262 nécessite l'extraction d'un certain nombre de métriques liées à la résilience du système face aux pannes simples et multiples, ainsi qu'à l'efficacité des contre-mesures (détection, signalement et correction des erreurs) et modes de défaillance. L'extraction des métriques d'échec à partir des arbres de défaillance est une méthode connue et documentée dans la littérature. Néanmoins, son application a souvent été limitée aux systèmes électromécaniques macroscopiques tels qu'une voiture, un actionneur ou des chaînes de capteurs. D'autre part, ces méthodes sont rarement appliquées dans le domaine des SoC automobiles où l'extraction des métriques est encore largement manuelle (généralement à l'aide d'un tableur) et dépendante d'un expert, et où la vérification de l'efficacité des contre-mesures est mieux effectuée par injection de fautes ciblée sur quelques sous-parties du système complet ou par irradiation sous un faisceau de particules. Cette thèse propose de développer une méthodologie d'extraction de métriques de fiabilité basée sur l'injection de fautes par bloc ainsi que des méthodes de composition pour obtenir les métriques au niveau du système complet. La première partie de la thèse sera consacrée à l'étude de la bibliographie sur la construction d'arbres de défaillance, la norme ISO26262 et la déclinaison des différentes métriques de fiabilité dans le cas d'un système SoCs numérique. L'extraction des métriques au niveau du bloc sera basée sur deux méthodes différentes, l'une analytique basée sur les probabilités, l'autre expérimentale basée sur l'injection de fautes. L'objectif n'est pas de développer de nouveaux

codes de probabilité ou des outils d'injection de fautes, mais de développer une méthodologie pour les utiliser dans le contexte d'un SoC afin d'obtenir les données souhaitées. La deuxième partie de la thèse concernera la composition des données obtenues au niveau du bloc fonctionnel afin d'obtenir les métriques ISO26262 au niveau du système (SoC). Il s'agira de développer une méthode de composition adaptée en particulier aux caractéristiques des SoCs (système de communication, effectuant des calculs devant réagir en temps réel, ...) et aux modèles de défaillance qui les caractérisent ou imposés par la norme ISO26262.

INDEX WORDS: [FMEA, ISO26262, System on Chip, Reliability, Functional Safety]

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**TIZIANO FIORUCCI**

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**TIZIANO FIORUCCI**

Major Professor: Giorgio di Natale  
Industrial Advisor: Jean Marc Daveau  
Committee:  
Alberto Bosio  
Stefano di Carlo  
Pascal Benoit  
Vincent Berouille

Electronic Version Approved

I am Me.

## DEDICA

La decisione di intraprendere un percorso di studi aggiuntivo a ciò' che la maggior parte degli ingegneri intraprende e' qualcosa su cui si siflette molto. Sono anni tolti alla carriera, anni di instabilità prolungata oltre a quello che normalmente ci si protrebbe aspettare. In fondo pero' la prolungazione dello status da studente piu' che per i normali anni del liceo, e per alcuni, gli anni dell'università, non e' un piacere che apprezzano in molti, ed e' impossibile ignorare il fatto che se questa distinzione esiste, e se si sente di far parte di quel tipo di personalità, puo' solo che essere un peccato ignorare che esiste una comunità di scienziati-ingegneri-medici-letterati che condivide l'amore per la propria materia allo stesso modo e merita quanto meno di essere esplorata e conosciuta, puntando eventualmente a unircisi e tentare di farne parte.

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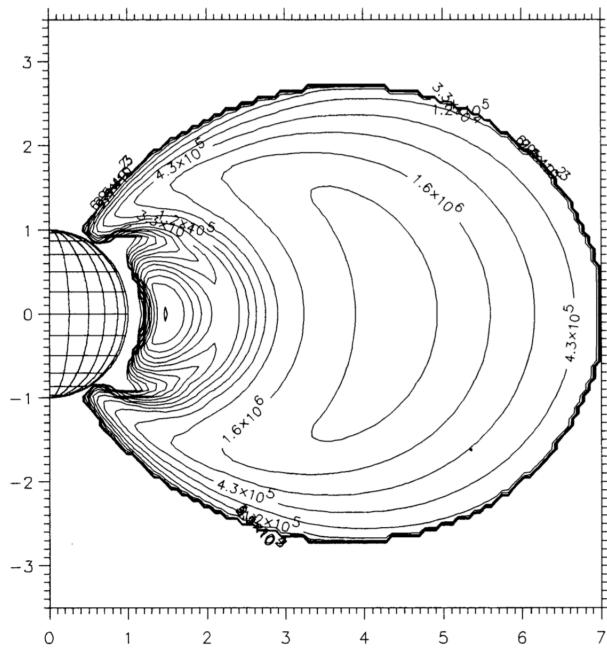
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# CHAPTER I

## PROBLEM AND ITS BACKGROUND

*Semiconductor devices and integrated circuits are nowadays operated in a number of hostile environments, therefore it worth to analyze all of them in order to determine what threat show up. Moreover in this chapter it will shown the most common effects on MOSFET based devices as well as the possible architectural solution of the state of the art*



## I.I The Space Environment

The Earth and its immediate surroundings are protected by the atmosphere, which acts as a semi-permeable shield letting through light and heat while stopping radiation and UV's; because no such protection is available in space, human beings and electronics (onboard Earth orbiting satellites, space shuttles, space probes) must be able to cope with the resulting set of constraints. Based on several tens of years of this space era, a detailed analysis of the problems on satellites shows that the part due to the radiation environment is significant. It appears that the malfunctions are due to problems linked to the space environment (9 to 21%), electronic problems (6 to 16%), design problems (11 to 25%), quality problems (1 to 8%), other problems (11 to 33%) and problems that are still unexplained (19 to 53%) [10]. It is clear that the unexplained problems are either problems linked to the space environment, to the electronics, to the design, or otherwise but the information collected on the ground is generally not sufficient to define the origin of the problem. The space environment is largely responsible for about 20% of the anomalies occurring on satellites and a better knowledge of that environment could only increase the average lifetime of space vehicles.

So the study of the space environment and its causes started, in all its great variety of environments depending on different orbital levels and electromagnetic forces involved. The degradations and disturbances induced by space radiation in the materials and the electronic components are phenomena that have been studied for many years.[10] It resulted in a basic classification of damages, either for Humans and for Electronics that can be easily divided into two groups each:

For Electronics

1. **Cumulative** such as the degradation of thermal control coatings, optics and electronics and the erosion of materials;
2. **Sporadic** such as noises in the detectors and optics, single event effects in highly integrated electronic circuits and electrostatic discharges.

while for humans

1. **Immediate**, permanent or delayed non stochastic effects (destruction or modification of cells), the speed with which the symptoms appear and their seriousness increase in proportion to the exposure to the radiation;
2. **Stochastic** associated with the modifications to the cells whose probability of appearing in the long term increases in proportion to the irradiation (cancers, leukemia, (SET) Program. genetic effects).

### 1.1.1 Solar Activity

Sun is either a source and modulator of space radiation, its activity can be described using a cyclical model. Each cycle has approximately 11 years long. In this time span the Sun has 7 years of maximum activity and 4 at its minimum, the transition is considered sharp even though it is indeed continuous. Moreover every 11 years cycle the Sun reverse its magnetic polarity, this leads to an actual 22 years period between two equal configurations.

Usually two main indicators are used to describe the solar activity:

1.  $F_{10.7}$  - 10.7 cm radiation flux
2. Sunspots count - The numbering of sunspot cycles began in 1749 and it is currently near the end of solar cycle 23. The record of  $F_{10.7}$  began part way through solar cycle 18 in the year 1947

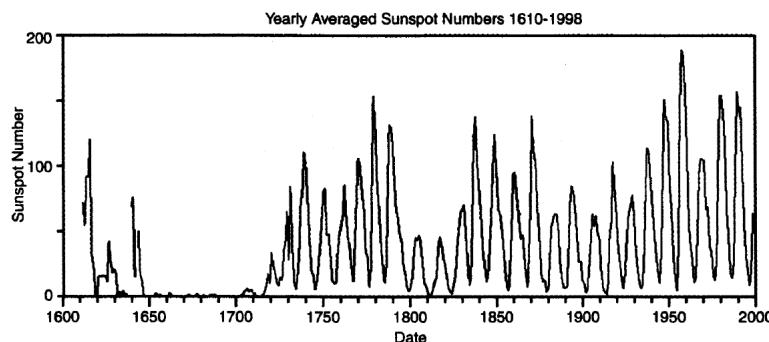


Figure 1.1: The observed record of yearly averaged sunspot numbers [10]

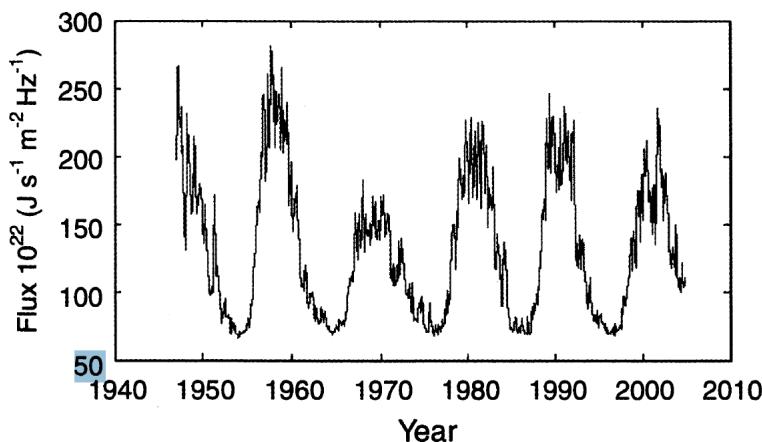


Figure 1.2: Measured values of solar 10.7 cm radio flux [10]

Large solar particle events are known to occur with greater frequency during the declining phase of solar maximum [3]. Trapped electron fluxes also tend to be higher during the declining phase [4]. Trapped proton fluxes in low earth orbit (LEO) reach their maximum during solar minimum but exactly when this peak is reached depends on the particular location [5]. Galactic cosmic ray fluxes are also at a maximum during solar minimum but in addition depend on the magnetic polarity of the sun [6].

## 1.1.2 Cosmic Rays

With Galactic Cosmic Rays (GCR) it is intended all those highly-charged particles that have been generated outside our solar system, even though their precise origin is unknown, scientific community believes that Supernovas explosions may be the first source. Some general characteristics of GCR are listed in the following table

Hadron Composition	Energy	Flux	Radiation Effects	Metric
87% Protons				
12% Alpha	up to $10^{11}$ GeV	$1 \text{ to } 10 \text{ cm}^{-2} \text{ s}^{-1}$	SEE	LET
1% Heavy Ions				

But it is possible to have a deeper look at the relative abundances in 1.3.

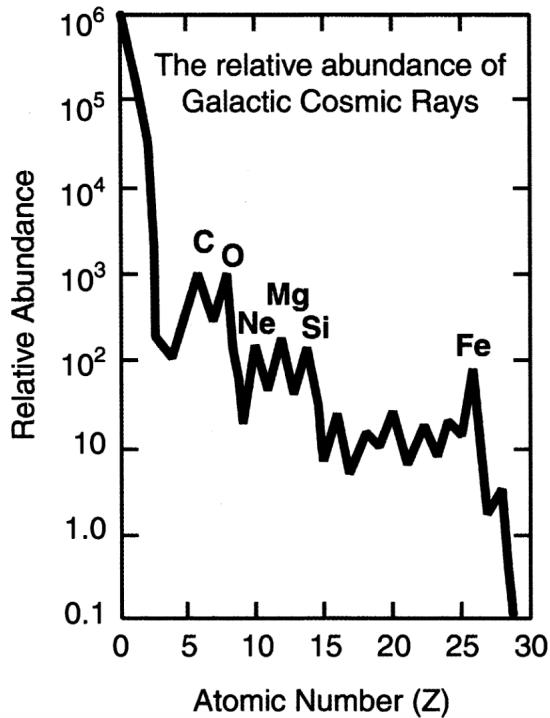


Figure 1.3: Abundances of GCR up through  $Z = 28$

All the elements in the Periodic Table up to Uranium are present in GCR although there is a steep drop-off for atomic numbers higher than iron ( $Z=26$ ). Their source isn't the only unknown feature of GCR. We saw that they can reach energies up to  $10^{11} \text{ GeV}$ , but the causes behind such acceleration are still to be comprehended. On the other hand their flux is really small, limited to a few  $\text{cm}^{-2} \text{s}^{-1}$ . Typical GCR energies and fluxes are shown in Fig. 1.4.

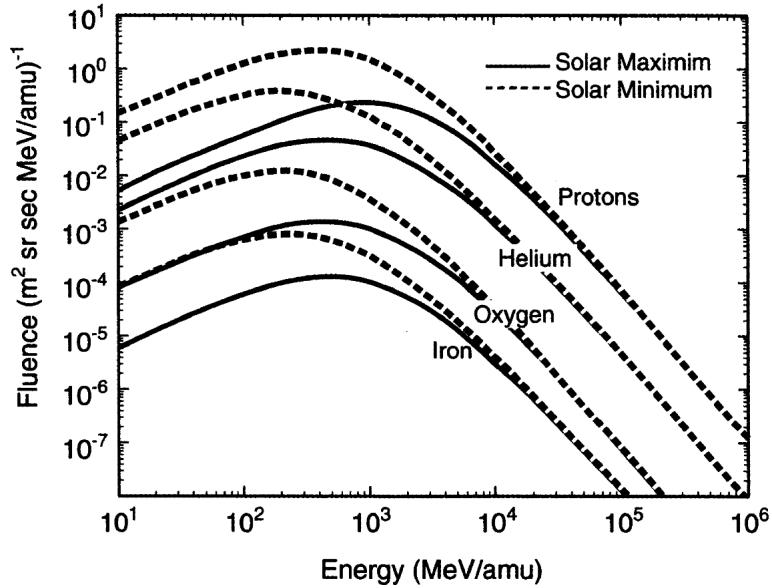


Figure 1.4: GCR energy spectra for protons, helium, oxygen and iron during solar maximum and solar minimum conditions

The peak around  $1\text{GeV}$  is due to the moderation effect of the solar wind and solar magnetic field, still another explanation for the inverse proportionality of GCR Flux and solar activity. Speaking about the Radiation Effects that these

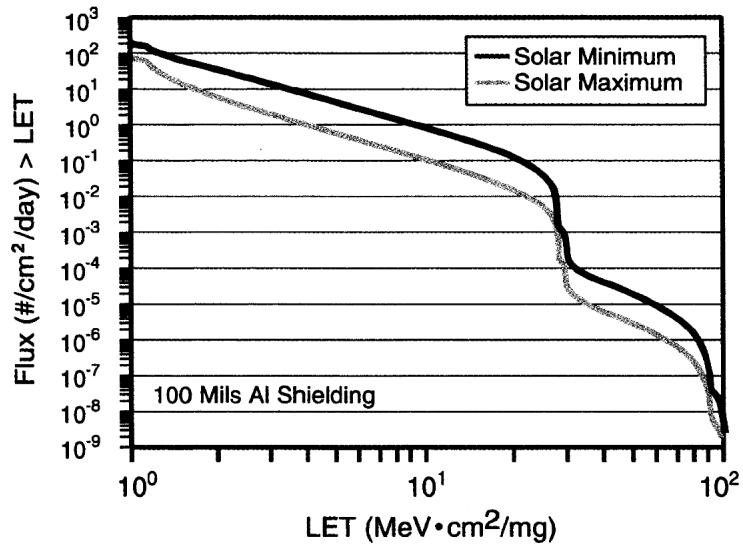


Figure 1.5: Integral LET spectra for GCR during solar maximum and solar minimum

GCR can cause, the main Result of impact is a Single Event Effects (SEE) and the metric to usually utilized to describe the heavy ion induced SEE is the Linear Energy Transfer (LET) which can be defined as the energy lost by the ionizing particle per unit of path lenght in the sensitive section of the device. So it is possible to convert Fig. 1.4 into the spectrum per LET, integrating we can see the difference between the minimum and maximum activity level of the Sun. In the following image all the contributes from all the elements, starting from protons up to uranium, have been considered. The ordinate gives the Flux of particles having a LET above the corresponding abscissa.

The LET Metric can be applied in GEO and Interplanetary missions, in absence of geomagnetic attenuation. For instance, due to basic interaction between charged particles and Earth magnetic field they tend to follow the geomagnetic lines and so parallel to the plant surface at the equator. Thus the energy is mostly deflected away. The Effect of the Geomagnetic field on the incident GCR-LET spectrum during solar minimum is discussed for various orbits in [12]

### 1.1.3 Radiation Belts

Earth is relatively well protected against external influences such as radiation coming from outer space. In these terms we can imagine the Earth Magnetosphere as the natural cavity in the interplanetary medium that serve to the cause. It is compressed on the solar side and highly extended on the anti-solar side. Poles represent the only space offered to the interplanetary particles to penetrate into the upper atmosphere. Meanwhile the charged particles close to Earth can be trapped by the magnetic field and form the Radiation Belts. As shown in Fig: 1.6 the radiation belts only occupy a limited internal region of the whole magnetosphere. Starting from the closest section to Earth it's possible to identify the upper atmosphere, constant over time. On the opposite end, we cannot really define a boundary due to its strong dependence on solar wind and magnetic field.

In the Earth Magnetosphere we can define the magnetic field as the sum of two contributes:

1. **Main Component** - This term is based on the convection motion in the core of the planet
2. **External Origin** - This includes all the permanent magnets of the terrestrial crust.

In a zero order approximation the field can be considered bipolar. However it is way more accurate to take an off-centered and tilted dipolar magnetic field

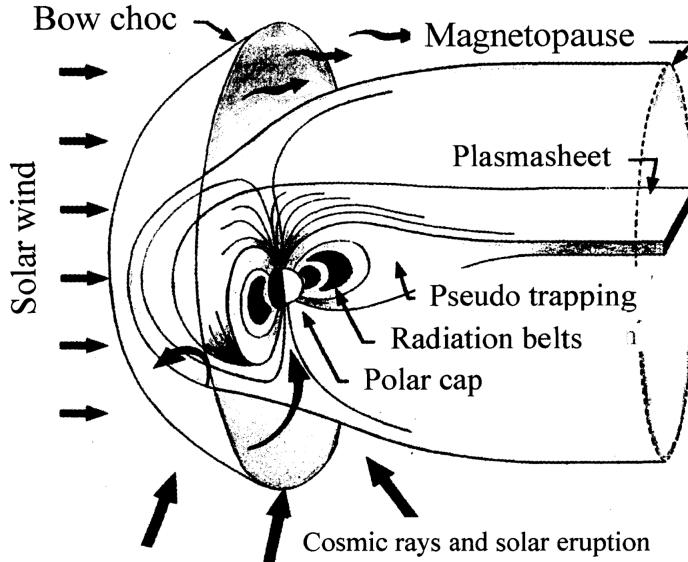


Figure 1.6: Magnetosphere with respect to Radiation Belts [10]

as approximation. This gives us a Dipole not centered in the center of Earth and having its axis not parallel to the earth one. This geometry leads to an anomaly in the magnetic field, a region in which the field is weaker, called the South Atlantic Anomaly, as shown in Fig. 1.7. It is important to observe that the magnetic field on Earth is evolving on a long term basis (secular drift), in particular the South Atlantic Anomaly, which is drifting south-eastwards. As the present time we note:

- a decrease in the intensity of  $27 \frac{nT}{year}$  ( $0.05\%$  a year) [10]
- a drift of the axis, resulting in a westward rotation of the southern end of the dipole ( $0.014\text{deg}$  a year) and an increase in the shift towards the West Pacific close to  $3\text{ km}$  a year. [10]

**Dynamics of the charged particles** In order to better understand and describe the dynamics of the charged particle in the magnetosphere, we can define a reference system and its coordinates.

- $r$  is the distance from the center of the dipole
- $\lambda$  is the latitude with  $\theta$  its colatitude ( $\theta = \frac{\pi}{2} - \lambda$ )
- $\phi$  its magnetic longitude

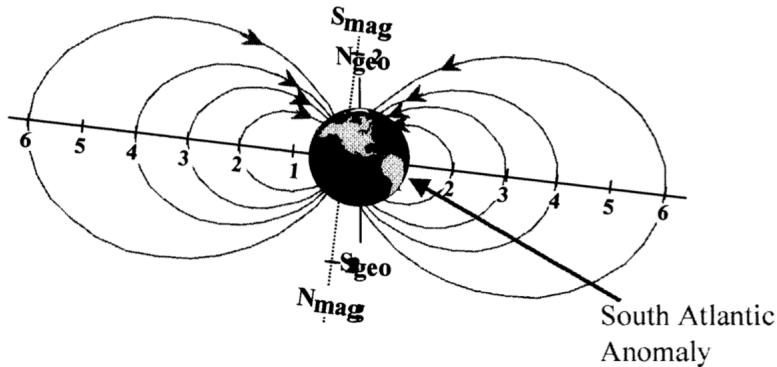


Figure 1.7: Dipolar magnetic field tilted and off-center with respect to Earth.  
[10]

Last we need to describe the Field lines or force line by the McIlwain parameter  $L$ , roughly equal to the distance from the center of the planet to the intersection point of that force line with the magnetic equatorial plane. So a single point in the field is called  $B$ , modulus of the magnetic field.

All charged particles subject to an electromagnetic field will be subject to the Lorentz force

$$F = q(v \wedge B + E) \quad (1.1)$$

Under these conditions, the movement of the high-energy particles can be generally broken down into three basic periodic movements.

**Gyration** All the charged particles in a magnetic field will rotate around the line of field. This is called Gyration and we can define this movement

1. the Larmor radius  $r_L = \frac{mv^2}{qB}$
2. the relativistic magnetic moment  $\mu = \frac{mv^2}{2B}$

**Bounce** If a particle only has a component of its velocity parallel to the magnetic field, then it will move along the field lines. In their motion they keep the magnetic moment  $\mu$  constant. Since the magnetic moment has to stay constant, while moving from the equator towards the poles, it is possible to notice a strongly increasing magnetic field. It is necessary that the perpendicular component of the speed should increase in order for  $p$  to remain constant.[10]

**Drift** in order to simplify the problem, we place ourselves on the magnetic equator. Since the magnetic field of the planets has a radial gradient, the gyration cannot take place in a constant Larmor radius. Indeed, the magnetic field along a gyration becomes stronger if the particle approaches the planet, the Larmor radius is then smaller and therefore the radius of the trajectory's curve is also smaller. The particle will thus be able to move away from the planet, the magnetic field will be weaker and therefore the Larmor radius and the radius of the trajectory's curve will be greater. The particle therefore does not go through a simple circle but along a more complex trajectory. This movement breaks down into a simple gyration (circular) and a rotation movement around the planet: this is the drift movement.

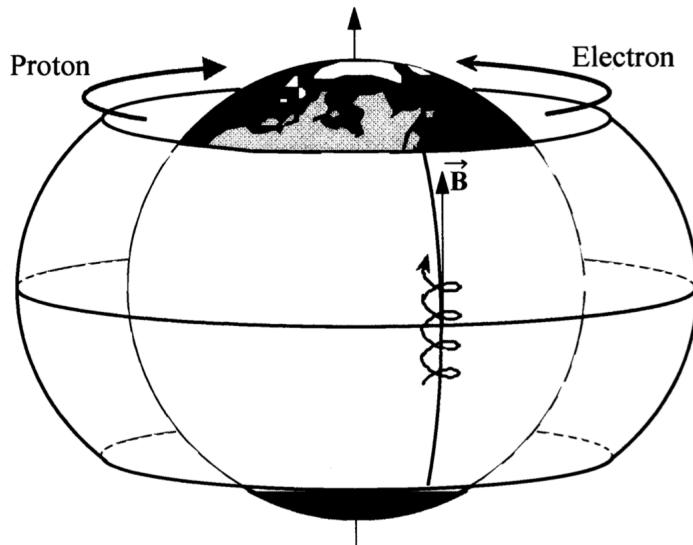


Figure 1.8: Composition of a charged particle's three periodic movements: gyration, bounce and drift. The particle then follows a torus surface called a drift shell. [10]

A charged particle submitted to these three basic [81] and periodic movements then moves through torus shaped surfaces around the Earth, which are commonly called drift shells Fig 1.8. The periods associated with each of these basic movements for a 3 MeV electron at  $L=3$  are respectively  $2.1410^{-4}$  s, 0.19 s and 504 s. The disparity between the periods is very great, a factor of the order of 1000 should be noted between each of them going from the gyration movement to the drift movement.

Due to the magnetic field in proximity of Earth make all the relativistic charged particles to remain trapped in a quasi periodic movement. These conditions are perfect to start an increasing high energy charged particles ensables,

creating the so called radiation Belts. Given the previously presented trajectories, the Radiation Belts assume a toroidal shape surrounding the Earth. The atmosphere is the lower bound while the outer limit is not well defined and may be time variable.

During the first space missions, J. Van Allen has discovered that mostly all the trapped particles are Protons and Electrons, having an Energy range between some KeV and hundreds of MeV. Below there is a representation of the Proton belt (Fig. 1.9), pretty stable and constant in time, with energies from some MeV to hundreds of MeV.

On the other hand, the electron belt is more complex (Fig. 1.10) and has two maximums respectively corresponding to the internal and external zones: - the first one centered on  $L = 1.4$  extends up to  $L = 2.8$ ; the electron populations are relatively stable there and can reach maximum energy levels of the order of 10 or even 30 MeV; - the second one, centered on  $L = 5$ , extends from  $L = 2.8$  to  $L = 10$ ; the electron flows there are much more variable and the energy levels can be as high as 7 MeV.

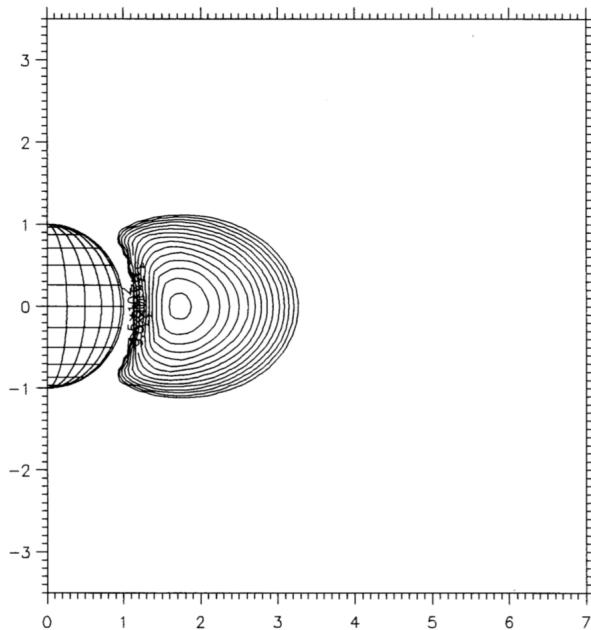


Figure 1.9: Proton radiation belt [10]

**Dynamic of the radiation Belts** Starting from the 1990s the American satellite CRRES has shown the extreme dynamics of protons and electrons trapped in the radiation belts. In fact the population of these particles is strongly dependent on two factors:

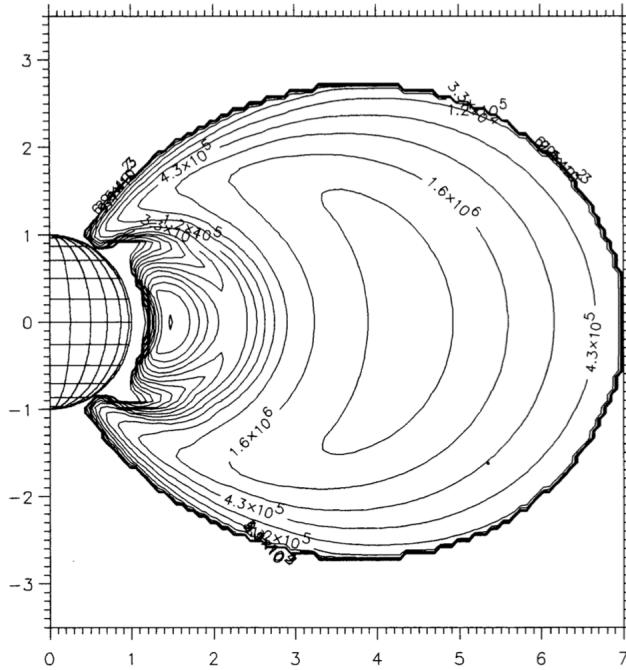


Figure 1.10: Electron radiation belt. [10]

1. The Sources - injections from the tail of the magnetosphere and creations by nuclear reactions
2. The Losses - Precipitations in the upper atmosphere or by charge exchange with particles from the Exosphere.

**Dynamics on the scale of the Solar cycle-Protons** The radiation belt of protons having high energies, more than 10 MeV varies slowly as function of the solar cycle, as shown in Fig:1.11. The flux levels oscillates around its maximum at the minimum activity of the Sun and vice versa. This is the actual result of two different phenomena, the absorbtion of the protons by the upper atmosphere and the modulation of CRAND (Cosmic Ray Albedo Neutron Decay) source. This balance is shown in Fig:1.11.

**Dynamics on the scale of the Solar Cycle-Electrons** As well as for the proton cycle, the electrons, especially in the geostationary orbit, follow a similar trend. Inversely proportional to the Sun activity cycle, as shown in Fig:1.12

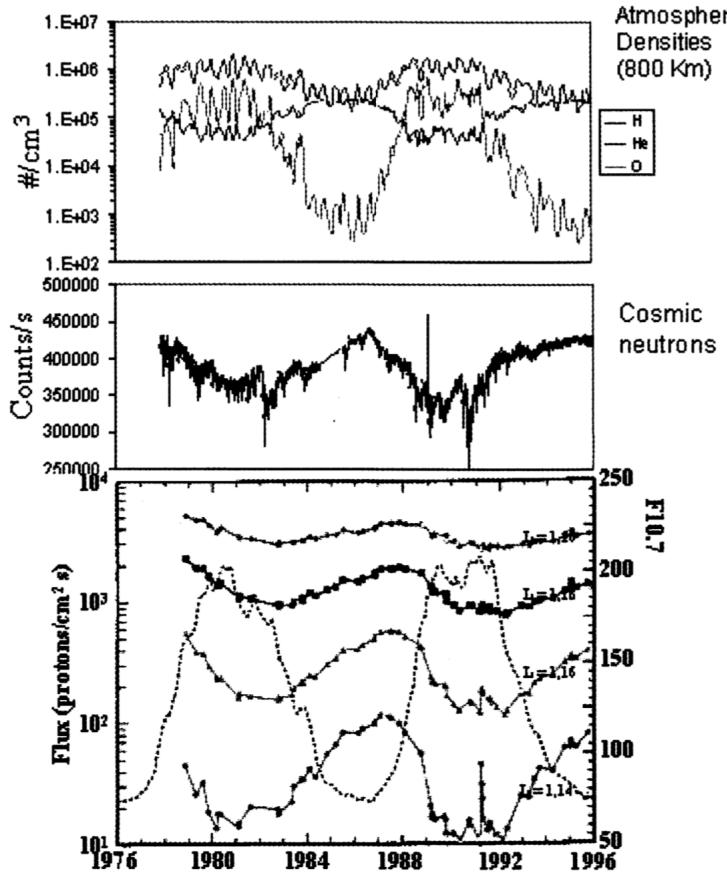


Figure 1.11: Changes in the proton fluxes at low altitudes (bottom), in the cosmic radiation (middle) and atmospheric densities (top) as a function of the solar cycle. [10]

## 1.2 The Earth Environment

Since the 1984, the existence and impact of atmospheric neutrons has been predicted. They can cause Single Event Upset in the electronics, the first actually measured was in 1992. After that several hundreds have been observed. As we can see in Fig. 1.13 Cosmic rays cover a large spectrum of energies, with a comparatively high flux in the 100 MeV to 10 GeV range and a peak around 500 MeV; cosmic particles collide with the nuclei of atoms making up the Earth atmosphere and initiate the so-called air showers, producing particles such as neutrons, protons, muons, pions, electrons and gamma-rays.

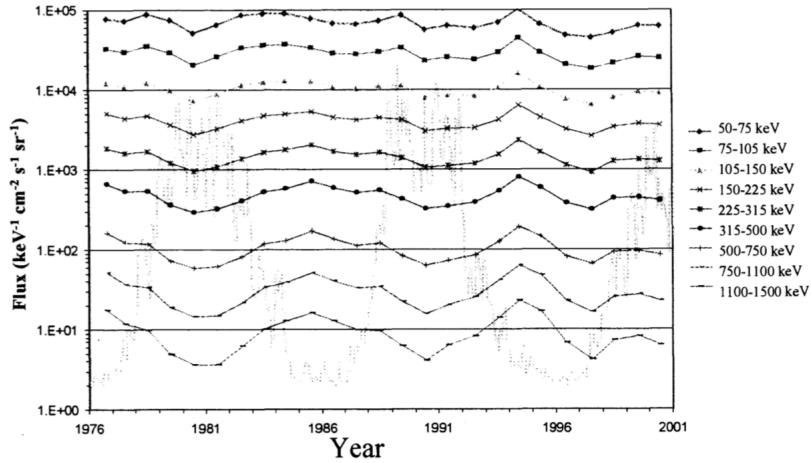


Figure 1.12: Electron fluxes at geostationary orbit as a function of the solar cycle.  
[10]

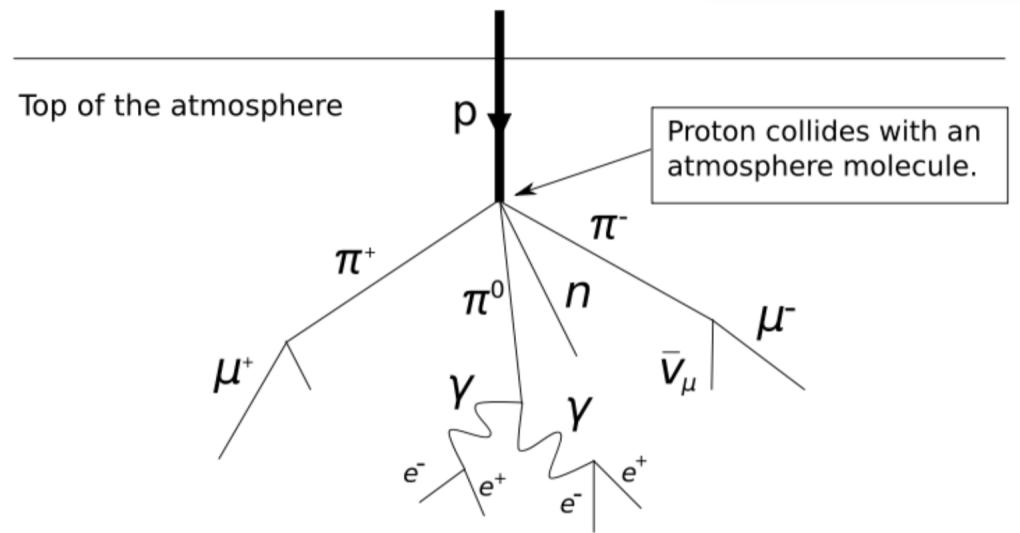


Figure 1.13: Example of Proton Scattering [40]

A deeper analysis of the particles at commercial flight level shows a great majority of neutrons, while protons play a minor role in the Single Event Upset at those altitudes.

Moreover, even though the scheme reported in Fig. 1.13 seems to be a top down shower, the neutron flux is, in fact, isotropic.

## 1.3 Military Environment

In case of an explosion occurring above the Earth atmosphere, two possible scenarios may be considered

- Aerospace systems operating at altitudes higher than 50-100 km will be directly submitted to the radiation emitted by the weapon; hard X-rays, gamma-rays and neutrons all have a significant impact on the electronic systems onboard satellites
- The main indirect effect to be considered is that related to trapping by the Earth magnetic field lines of electrons from fission debris, resulting in the formation of highly stable, artificial radiation belts which can deliver much higher radiation doses to satellites than natural belts; the first satellite failure due to radiation effects dates back to 1963, to the Starfish test (a 1.4 Mton thermonuclear bomb detonated at an altitude of 400 km); the test produced an intense radiation belt destroying seven satellites over seven months, primarily because of dose effects on their solar panels; the TELSTAR satellite, launched on July 10, 1962, broke down in February 1963

A nuclear explosion in the Earth atmosphere, besides a variety of mechanical effects, is responsible for different purely radioactive effects, which can be split into two main categories

- **Initial nuclear radiation (INR)** is that released within less than a minute after detonation; X-rays are quickly stopped (at about 500 m above sea) so that gamma-rays (responsible for ionizing dose effects) and neutrons remain
- **Residual nuclear radiation (RNR)** features several radiation sources; fission products, radioactivity of debris (neutron-activated weapon materials), that of unfissioned uranium and/or plutonium and activation of the environment; a further distinction can be made between local fallout, occurring less than 24 hours after explosion, with a resulting significant level of ground radiation, and worldwide fallout, which can take place at significant distances from the place of explosion

## **1.4 Radiation Effects on COTS Components**

### **1.4.1 Basic Damage Mechanism in Semiconductor Devices**

Even if we take into consideration the great diversity of particles and their interaction with different technologies, it is possible to distinguish just two main categories of damaging mechanism for semiconductor devices

1. Ionization Damage
2. Displacement Damage

#### **Ionization Damage**

The ionization takes place when energy deposited in a semiconductor or in insulating layers, chiefly SiO<sub>2</sub>, frees charge carriers (electron-hole pairs), which diffuse or drift to other locations where they may get trapped, leading to unintended concentrations of charge and parasitic fields; this kind of damage is the primary effect of exposure to X- and gamma-rays and charged particles; it affects mainly devices based on surface conduction (e.g. MOSFETs)[40]. Directly ionizing radiation consists of charged particles. Such particles include energetic electrons (sometimes called negatrons), positrons, protons, alpha particles, charged mesons, muons and heavy ions (ionized atoms). This type of ionizing radiation interacts with matter primarily through the Coulomb force, repelling or attracting electrons from atoms and molecules by virtue of their charges. Indirectly ionizing radiation consists of uncharged particles. The most common kinds of indirectly ionizing radiation are photons above 10 keV (x rays and gamma rays) and all neutrons. X-ray and gamma-ray photons interact with matter and cause ionization in at least three different ways:

- Lower-energy photons interact mostly via the photoelectric effect, in which the photon gives all of its energy to an electron, which then leaves the atom or molecule. The photon disappears.
- Intermediate-energy photons mostly interact through the Compton effect, in which the photon and an electron essentially collide as particles. The photon continues in a new direction with reduced energy while the released electron goes off with the remainder of the incoming energy (less the electron's binding energy to the atom or molecule).
- Pair production is possible only for photons with energy in excess of 1.02 MeV. However, near 1.02 MeV, the Compton effect still dominates: pair

production dominates at higher energies. The photon disappears and an electron-positron pair appears in its place (this occurs only in the vicinity of a nucleus because of conservation of momentum and energy considerations). The total kinetic energy of the electron-positron pair is equal to the energy of the photon less the sum of the rest-mass energies of the electron and positron (1.02 MeV). These energetic electrons and positrons then proceed as directly ionizing radiation. As it loses kinetic energy, a positron will eventually encounter an electron, and the particles will annihilate each other. Two (usually) 0.511 MeV photons are then emitted from the annihilation site at 180 degrees from each other. For a given photon any of these can occur, except that pair production is possible only for photons with energy greater than 1.022 MeV. The energy of the photon and the material with which it interacts determine which interaction is the most likely to occur[55].

About indirect ionization, we can say that light particles such as protons and neutrons could have not enough LET to cause upset, but they can cause nuclear reactions that in turn create heavier particles that can cause upsets by direct ionization. Secondary reaction products have much higher LET but shorter ranges and lower energies[56].

### **Displacement Damage**

Incident radiation dislodges atoms from their lattice site, the resulting defects altering the electronic properties of the crystal; this is the primary mechanism of device degradation for high energy neutron irradiation, although a certain amount of atomic displacement may be determined by charged particles (including Compton secondary electrons); Displacement Damage mainly affects devices based on bulk conduction (e.g. BJTs, diodes, JFETs) [40]. Displacement damage occurs when sufficient energy is transferred from an incident energetic particle to a lattice atom to dislodge it from its normal location. Using Si as an example, the Si atom initially displaced by an incoming particle is known as the primary knock-on atom (PKA) or the primary recoil. The PKA, or recoil, carries a net charge that depends on its kinetic energy. Displacement damage occurs through the interaction of incident particles with Si atoms by any of the following three processes:

- Rutherford (i.e., Coulomb) scattering
- Nuclear elastic scattering
- Nuclear inelastic scattering

Once any of those basic interaction processes produces a PKA, that ion subsequently can introduce further displacement damage by Rutherford and nuclear scattering. Lattice defects are produced by PKAs and any later-generation energetic recoils that they create. When defects produced by incident radiation are relatively far apart, they are known as isolated, or point, defects. As an example, isolated defects are created by 1 MeV electrons incident on Si. Radiation-induced defects may also be created closely together and form local regions of disorder known as defect clusters. For example, incident 1 MeV neutrons produce both isolated and clustered defects in Si. In general, energetic particles incident on semiconductors create either isolated and clustered defects or solely isolated defects, depending on the mass and energy of the incident particles. Nearly all the effects of displacement damage on the electrical and optical properties of semiconductor materials and devices can be understood in terms of energy levels introduced in the bandgap. Those radiation-induced levels result in the following effects: recombination lifetime and diffusion length are reduced; generation lifetime decreases; majority and minority-carrier trapping increase; majority carrier concentration changes; thermal generation of electron-hole pairs is enhanced in the presence of a sufficiently high electric field; tunnelling at junctions is enabled. In addition, radiation-induced defects reduce the carrier mobility and can exhibit metastable configurations[**biblio**].

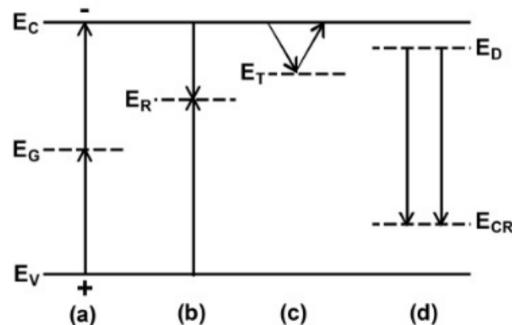


Figure 1.14: Radiation-Induced energy

Figure 1.14 illustrates radiation-induced energy levels in the Si bandgap that give rise to the following processes: (a) enhanced thermal generation; (b) enhanced recombination; (c) enhanced temporary trapping; (d) reduced carrier concentration due, in this example, to the introduction of centers that compensate for donors (carrier removal).

## **I.4.2 Radiation Effects in Electronics**

As previously stated, the effects of radiation on semiconductor devices can be divided into two broad classes. Previously the Total Dose effects have been explained, now the focus will be moved to Single Event Effects.

### **Single Event Effect**

These effects are due to the deposition of charge by a single particle that goes through a sensitive region of the device. This can lead to a destructive or non destructive damage of the device. Moreover it is possible to identify a couple of differences between Total Dose effects and Single Event Effects:

- Single Event Effects are Stochastic, while TID effects are cumulative and may occur after the device has been exposed to radiation for a long time
- TID are related to Long Term response, while SEE to Short term Response
- Only a very limited portion of the device is affected by SEE, while TID affects uniformly the entire device, this is due to the number of particles hitting the device and their distribution.
- While for TID the main figure is the drift of the main device parameters, concerning SEE the most important figure is the Rate of Occurrence.

We can so define this effects some of the SEE as "Soft" in case they do not induce any physical damage to the device, but just an information loss. Otherwise they are categorized as "hard" in case the impact of a heavy ion is followed by the rupture of the gate oxide. Here there is a list of the major Soft effects and Hard effects

### **Main Classes of Soft Effects**

- Single Event Upset (SEU) - the corruption of a single bit in a memory array
- Multiple Bit Upset (MBU) - the corruption of multiple bits due to a single particle
- Single Event Transient (SET) - a transient signal induced by an ionizing particle in a combinatorial or analog part of a circuit

## Main classes of hard effects

- Single Event Gate Rupture (SEGR) - rupture of gate oxide occurring especially in power MOSFETs
- Single Event Burnout (SEB) - burnout of a power device
- Single Event Latch-Up (SEL) - the activation of parasitic bipolar structures, leading to a sudden increase of the supply current

Usually in order to evaluate the occurrence of SEE the cross section of the device is used. It is defined as follow:

$$\sigma_{SEE} = \frac{Number\,Of\,Events}{Particle\,Fluence} \quad (1.2)$$

the Cross Sections varies as function of the LET of the particle hitting the device, observable only if higher than the threshold LET. The charged released by the particle hitting the transistor is collected via the so called Funneling mechanism. Most of the charge is sucked in at the struck junction through a deformation of the junction potential, while the remaining charge diffuses in the substrate and may be collected or not at the same junction[40]

The aim of these thesis is to focus on Soft Effects, in the next section the most common SEE will be analyzed in details.

## Radiation Effects in MOSFETs

**Single Event Upset** It is obvious that in order to cause disturbance in any circuit the charge generated by a particle hitting the device must be in a sensitive node; in particular reverse biased PN junctions are the most affected by collected charge, having a larger deplation reagion and stronger electric field. Taking into account the case of an SRAM cell, may be the case of a particle hitting the Drain of the off NMOSFET. If that is the case the released charge is collected by the reverse-biased drain, the voltage at the struck node tends to decrease, turning the radiation-induced current in to a voltage transient. The current decreases the potential at the node, and it may go as low as below the switching voltage, changing the initial state.

These effects are function of the LET of the impinging particle and the incident angle  $\theta$

**Multiple Bit Upset** Single events effects have become more complex to study as the new technologies are released. In particular the minimum lentgh that can be obtained while creating CMOS devices in lithography has gone below the

micron realm. Nowadays the size of the path of the hitting particle has become comparable to the size of modern chips. Therefore that in the past may have involved a single point in the circuit now involve multiple nodes and charge sharing may occur. It follows that the rate of occurrence of Multiple Bit Upset is strongly bound to rise as fabrication process evolve.

**Radiation Induce Latch-Up** A latch-up is a type of short circuit which can occur in an integrated circuit. More specifically it is the inadvertent creation of a low-impedance path between the power supply rails of a MOSFET circuit, triggering a parasitic structure which disrupts proper functioning of the part, possibly even leading to its destruction due to overcurrent. A power cycle is required to correct this situation.

**Single event latch-up is a latch-up caused by a single event upset, typically heavy ions or protons from cosmic rays or solar flares.** The parasitic

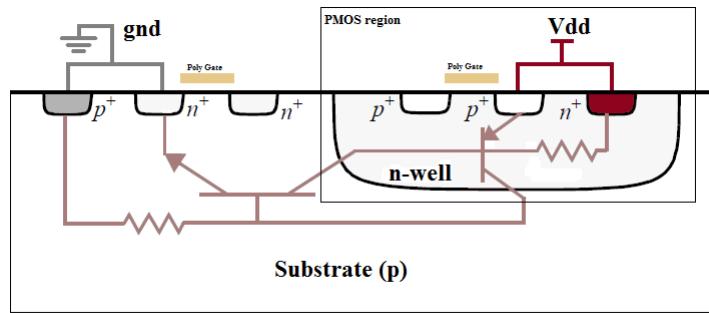


Figure 1.15: Latch-up Diagram

structure is usually equivalent to a thyristor (or SCR), a PNPN structure which acts as a PNP and an NPN transistor stacked next to each other. During a latch-up when one of the transistors is conducting, the other one begins conducting too. They both keep each other in saturation for as long as the structure is forward-biased and some current flows through it - which usually means until a power-down. The SCR parasitic structure is formed as a part of the totem-pole PMOS and NMOS transistor pair on the output drivers of the gates.

## 1.5 Types of redundant architectures

While there are various methods to implement redundant architectures, techniques and terminologies, the following section wants to represent the most common ones used in the industry.

### 1.5.1 Standby Redundancy

Standby redundancy, also known as Backup Redundancy is when you have an identical secondary unit to back up the primary unit. The secondary unit typically does not monitor the system, but is there just as a spare. The standby unit is not usually kept in sync with the primary unit, so it must reconcile its input and output signals on takeover of the Device Under Control (DUC). This approach does lend itself to give a "bump" on transfer, meaning the secondary may send control signals to the DUC that are not in sync with the last control signals that came from the primary unit. You also need a third party to be the watchdog, which monitors the system to decide when a switchover condition is met and command the system to switch control to the standby unit and a voter, which is the component that decides when to switch over and which unit is given control of the DUC. The system cost increase for this type of redundancy is usually about 2X or less depending on your software development costs. In Standby redundancy there are two basic types, Cold Standby and Hot Standby. [bib12]

#### Cold Standby Redundancy

In cold standby, the secondary unit is powered off, this is preserving the reliability of the unit. The drawback with respect to the hot standby is the longer downtime needed to switch from one unit to the secondary one. this makes it more challenging from the synchronization issues point of view.

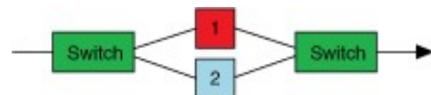


Figure 1.16: Cold Standby scheme

#### Hot Standby Redundancy

In hot standby instead, the secondary unit is always powered on and can eventually monitor the DUC. If the secondary unit is used as watchdog or voter to decide when to switch over, it is possible to eliminate the need for a third party unit to perform these operations. It is also possible to notice that some versions of the Hot standby are similar to the Dual Modular Redundancy (DMR) or Parallel Redundancy.

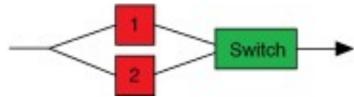


Figure 1.17: Hot Standby Schema

### 1.5.2 N-Modular Redundancy

N Modular Redundancy, also known as Parallel Redundancy, refers to the approach of having multiple units running in parallel. All units are highly synchronized and receive the same input information at the same time. Their output values are then compared and a voter decides which output values should be used. This model easily provides bumpless switchovers. This model typically has faster switchover times than Hot Standby models, thus the system availability is very high, but because all the units are powered up and actively engaged with the DUC, the system is at more risk of encountering a common mode failure across all the units.

Deciding which unit is correct can be challenging if you only have two units. Sometimes you just have to choose which one you are going to trust the most and it can get complicated. If you have more than two units the problem is simpler, usually the majority wins or the two that agree win. In N Modular Redundancy, there are three main typologies: Dual Modular Redundancy, Triple Modular Redundancy, and Quadruple Redundancy.

#### Dual Modular Redundancy

Dual Modular Redundancy or DMR uses two identical and so functional equivalent units, both of them able to control the DUC. The most challenging side of this configuration is the switching decision between the two units. Since both of them are monitoring the DUC there is the need for a routine in case of mismatch between the two units. It is possible to create a tiebreaker or even designate the second as default winner, assuming it is more trustworthy than the primary unit. The average cost increase of a DMR system is about twice that of a non-redundant system, factoring in the cost of the additional hardware and the extra software development time.

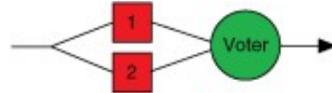


Figure 1.18: DMR scheme

### **Triple Modular Redundancy**

Triple Modular Redundancy (TMR) uses three functionally equivalent units to provide redundant backup. This approach is very common in aerospace applications where the cost of failure is extremely high.

TMR is more reliable than DMR due to two main features. The most immediate is that there are two "standby" units instead of a single one. The second is that in TMR it is common to see the so called diversity platforms or diversity programming techniques applied. In these techniques it is possible to notice the use of different hardware or software platforms.

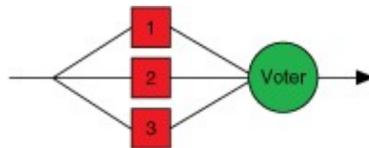


Figure 1.19: TMR Schema

### **Quadruple**

Quadruple Modular Redundancy (QMR) is fundamentally similar to TMR but using four units instead of three to increase the reliability. The obvious drawback is the 4X increase in system cost.

### **1:N Redundancy**

This design technique is used in case the system has a single backup for multiple modules and this backup is able to act as any of the single ones. This technique offers a redundancy at much lower costs than the others. This approach only works well when the primary units all have very similar functions, thus allowing the standby to back up any of the primary units if one of them fails.

Other drawbacks of this approach are the added complexity of deciding when to switch and of a switch matrix that can reroute the signals correctly and efficiently.

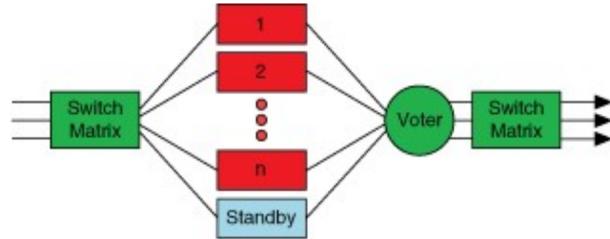


Figure 1.20: Example of Proton Scattering [40]

#### 1.5.4 Redundancy Improves Reliability

Reliability is defined as the probability of not failing in a particular environment for a particular mission time. Reliability is a statistical probability and there are no absolutes or guarantees. The goal is to increase the odds of success as much as you can within reason.

The following equation is the most common to calculate reliability, and it assumes that the system has a constant failure rate  $\lambda$

$$R(t) = e^{-\lambda t} \quad (1.3)$$

in which:

- $R(t)$  is the probability of success
- $t$  is the mission time or the time the system has to execute without an outage
- $\lambda$  is the constant failure rate over time ( $N$  Failures per hour)
- $\frac{1}{\lambda}$  is the MTTF or mean time to failure

In fact one way to calculate reliability is to take the probability equation and instead solve for the mean time to failure (MTTF) of the system:

$$R(t) = e^{-\lambda t} = e^{-\frac{t}{MTTF}} \quad (1.4)$$

solving for MTTF

$$MTTF = - \left( \frac{t}{\ln [R(t)]} \right) \quad (1.5)$$

For example, if your application had a mission time of 24 hours a day, 7 days a week, for one year (24/7/365) and you experienced a success rate of 90%:

$$MTTF = - \left( \frac{1\text{yr}}{\ln[.90]} \right) = 9.49\text{years} \quad (1.6)$$

in case redundancy has been added to the system we can, for instance, increase the success rate to 99% for the same mission time and therefore:

$$MTTF = - \left( \frac{1\text{yr}}{\ln[.99]} \right) = 99.50\text{years} \quad (1.7)$$

These equations effectively demonstrate the vast improvement in reliability that redundancy can bring to any system.

In particular, it is useful for this thesis to better understand the real advantages of TMR over the Simplex model. First we have to set some assumptions:

1. TMR only works if there are at least 2 working modules
2.  $R_m$  is the Reliability of the single module
3.  $R_v$  is the Reliability of the Voter

That said, it is possible to calculate the Reliability of a TMR system as follow:

$$R_{TMR} = R_v \sum_{i=2}^3 \binom{3}{i} R_m^i (1 - R_m)^{3-i} \quad (1.8)$$

and so

$$R_v [R_m^3 + 3R_m^2(1 - R_m)] = R_v (3R_m^2 - 2R_m^3) \quad (1.9)$$

from which it is possible to evaluate the MTTF for the same system as:

$$MTTF_{TMR} = \int_0^\infty R_{TMR} dt = \int_0^\infty R_v (3R_m^2 - 2R_m^3) dt \quad (1.10)$$

$$\int_0^\infty e^{-\lambda_v t} (3e^{-2\lambda_m t} - 2e^{-3\lambda_m t}) dt = \frac{3}{2\lambda_m + \lambda_v} - \frac{3}{3\lambda_m + \lambda_v} \quad (1.11)$$

It is possible to neglect the failure rate of the voter since it is usually designed to be way lower than the module one.

Now comparing the  $MTTF_{TMR}$  with the Simplex solution we obtain:

$$MTTF_{TMR} = \frac{3}{2\lambda_m} - \frac{2}{3\lambda_m} = \left(\frac{5}{6}\right) \left(\frac{1}{\lambda_m}\right) = \frac{5}{6} MTTF_{Simplex} \quad (1.12)$$

# CHAPTER 2

## FAILURE MODE AND EFFECT ANALYSIS FMEA

### 2.1 Introduction

#### 2.1.1 General

FMEA and FMECA are important techniques for a reliability assurance programme. They can be applied to a wide range of problems which may occur in technical systems, and can be carried out in varying degrees of depth, or modified, to suit a particular purpose. The analysis is carried out in a limited way during the conception, planning, and definition phases and more fully in the design and development phase. It is however important to remember that the FMEA is only part of a reliability and maintainability programme which requires many different tasks and activities. FMEA is an inductive method of performing a qualitative system reliability or safety analysis from a low to a high level. A thorough understanding of the system under analysis is essential prior to undertaking FMEA. Functional diagrams and other system drawings are normally necessary for this understanding. Reliability block diagrams, fault trees and/or state diagrams are then usually derived from these in order to carry out the analysis. In many instances the block diagram descriptions and block diagram failure descriptions are included in the FMEA format. Separate diagrams will be needed for the following:

1. The way in which different criteria for system failure are determined;
2. Degradation of function or reduction in assurance of function;
3. Alternative operational phases

## **2.1.2 Purpose of the Analysis**

The reasons for undertaking FMEA (or FMECA) may include the following:

- to identify those failures which have unwanted effects on system operation, e.g. safety critical failures;
- to satisfy contractual conditions that an FMEA should be completed;
- where appropriate, to quantify the reliability and/or safety of the system;
- to allow improvements of the system's reliability and/or safety (e.g. by design or quality assurance action)
- to produce aids to fault diagnosis;
- to allow improvement of the system's maintainability (by highlighting areas of risk or non-conformance for maintainability).

In view of these reasons the objectives of an FMEA (or FMECA) may include the following:

1. a comprehensive identification and evaluation of all the unwanted effects within the defined boundaries of the system being analysed, and the sequences of events brought about by each identified item failure mode, from whatever cause, at various levels of the system's functional hierarchy;
2. the determination of the significance (or criticality) of each failure mode with respect to the system's correct function or performance and the impact on the reliability and/or safety of the process concerned;
3. a classification of identified failure modes according to relevant characteristics, including detectability, diagnosability, testability, item replaceability, compensating and operating provisions (repair, maintenance, logistics, etc.);
4. an estimation of measures of the significance and probability of failure.

## **2.1.3 Basic Principles of FMEA**

The following concepts are essential to FMEA:

1. breakdown of the system into 'elements';

2. a diagram of the system's functional structure and identification of the various data which are needed to perform the FMEA;
3. the failure mode concept (a part may have several failure modes or a failure mode may involve several parts);
4. identification of new physical features or new requirements;
5. the criticality concept and the measure to be used (if criticality analysis is required).

Further, it is essential to specify the existing links between the FMEA (and the FMECA) and other qualitative (and quantitative) analytical methods within the overall reliability programme. Very few designs are wholly new. Most are to some extent developments of old designs. FMEA should use the information on existing systems and draw attention to the need for tests, etc. for the new parts.

## **2.2 Procedure**

### **2.2.1 General**

The wide variation in complexity of system designs and applications may require the development of highly individualized FMEA procedures consistent with the information available. Traditionally, there have been wide variations in the manner in which FMEA is conducted and presented. However, the analysis is usually done in a standard manner and presented on a worksheet that contains a core of essential information which can be developed and extended to suit the particular system or project to which it is applied. A typical example of a worksheet is shown in Figure 1.

The procedure consists of the following four main stages:

1. Preparatory definition of the system including the design, functional, operational, maintenance, and environmental requirements;
2. Establishment of the basic principles and purposes of the FMEA and the form of its presentation;
3. Carrying out the FMEA using the appropriate worksheet designed according to (a) and (b);
4. Reporting of the complete analysis including any conclusions and recommendations made.

A more detailed consideration of the information needed is given in Section 2.4.2.

### **2.2.2 Preparation**

At the commencement of an analysis, the following preparations should be made:

1. The analyst should have available the information listed in Section 2.4.2.2 to 2.4.2.7 that clearly defines the system to be analyzed.
2. It will usually be necessary for the analyst to translate the information into some form of functional, hierarchical, or reliability block diagrams. An example of a functional diagram is shown in Figure 2. This diagram shows how the failure effects at the part level form the failure modes at the module level, the failure effects at the module level form the failure modes at the subsystem level, and so on. Such a representation of the system should explicitly identify the system's functional structure, the system boundary, and the inputs and outputs crossing that boundary. Further information is given in Section 2.4.2.8 to 2.4.2.10.

### **2.2.3 FMEA principles**

The following principles should be applied:

1. Define clearly the purposes and uses of the FMEA as indicated in Section 2.1.2.
2. Establish and define the relationships with other forms of reliability analysis with which the FMEA may subsequently be integrated. (See Section 2.3.5.)
3. Define the scope of the FMEA in relation to the functional structure and hierarchical structure of the system as described by the block diagrams referred to in Section 2.4.2.10. It is essential to define the lowest level in the system's hierarchical structure at which the analysis will start. The guidance given in Sections 2.3.4, 2.4.1, and 2.4.2.8 is especially important for this task.
4. Define the format of the FMEA worksheet to suit the project requirements. The core information considered essential is as follows:
  1. The name of the item in the system being analyzed;

2. Function performed by the item;
3. Identification number of the item;
4. Failure modes of the item;
5. Failure causes;
6. Failure effects on the system;
7. Failure detection methods;
8. Compensating provisions;
9. Severity of effects;
10. Remarks.

Other information required for the particular system and project needs to be defined by the analyst according to the purposes of the

### **2.3 Analysis**

It is worth to underline that, even though the scope of this thesis is to establish a new methodology for automated FMEA in both hardware and software systems, there is the absolute need to comply with what is today the state of the art for FMEA and the guidelines to be followed to bring the metrics extracted during the analysis to certification. It is for this reason that the following section will present the traditional method applied today to all the electromechanical systems under evaluation. Bare in mind that the totality of the procedure described below is man driven.

The usual requirement and purpose of an FMEA is to identify the effect of all failure modes of all constituent items at the lowest level in the system. To achieve this the worksheet should be used in the following manner:

1. Identify all items in the system or subsystem, each of which is to have its failure modes and effects analysed. The system of identification by name and number should be such that no items will be omitted.
2. Select the first item for analysis and enter the item name and identification number in the appropriate columns of the worksheet. Determine the function of that item in the system and enter that on the worksheet.

3. Deduce all the possible failure modes of the item due to any possible cause and individually enter these modes on the worksheet
4. Postulate the most likely failure causes for each failure mode of the item and enter these on the worksheet. It will usually not be possible to consider all possible causes because the range is so vast, but the most significant with regard to the item, the failure mode and the application should be identified.
5. Deduce the effects of the failure on the subsystem and system, as determined by the scope of the FMEA
6. Complete the remaining columns of the worksheet for the first failure mode of the first item.
7. Repeat 3 to 5 for all failure modes of the first item
8. Repeat 2 to 6 for all other items

### **2.3.1 Multiple Stages**

If the FMEA is to be done in stages that each relate to separate levels in the system's hierarchical structure, the failure effects from the lower level become the failure modes at the next level up. The analysis should then proceed as follows.

1. Identify the lower level FMEAs that are appropriate for the next stage in the system FMEA according to the system's hierarchical structure defined by the block or functional diagrams (see 2.2, 2(b)). Where appropriate also include items defined as being at the lowest level in that part of the system structure
2. Perform the FMEA for each failure of each item at this higher level in the system structure as given in the previous section.
3. repeat the two above steps for any further higher levels in the system structure.

### **2.3.2 Worksheet recommendations**

The last worksheet entry should give any pertinent remarks to clarify other entries. Possible future actions such as recommendations for design improvements may be recorded and then amplified in the report. This column may also include the following:

- (a) any unusual conditions;
- (b) effects of redundant element failures;
- (c) recognition of specially critical design features;
- (d) any remarks to amplify the entry;
- (e) references to other entries for sequential failure analysis;
- (f) significant maintenance requirements;
- (g) dominant failure causes;
- (h) dominant failure effects;
- (i) decisions taken, e.g. at design review.

The report on the FMEA (or FMECA) may be included in a wider study or may stand alone. In neither case, the report should include a summary and a detailed record of the analysis and the block or functional diagrams which define the system structure. The report should also contain a list of the drawings (including issue status) on which the FMEA is based.

The summary should contain a brief description of the method of analysis and the level to which it was conducted, the assumptions and the ground rules. In addition, it should include listings of the following:

1. recommendations for the attention of designers, maintenance staff, planners, and users;
2. failures which, when initially occurring alone, result in serious effects;
3. failures which have no effect;
4. design changes which have already been incorporated as a result of the FMEA (or FMECA).

### **2.3.3 Report on Analisys**

The report on Failure Modes and Effects Analysis (FMEA) or Failure Modes, Effects and Criticality Analysis (FMECA) can be either a standalone document or a part of a broader study. In either case, the report should contain a summary and a detailed record of the analysis along with block or functional diagrams

Indenture level:			Design by:			Prepared by:						
Sheet no:			Item:			Approved by:						
Mission phase:			Issue:			Date:						
Item ref.	Item description-Function	Failure entry code	Failure mode	Possible failure causes	Symptom detected by	Local effect	Effect on unit output	Compensating provision against failure	Severity class	Failure rate (P/Mhr)	Data source	Recommendations and actions taken

Figure 2.1: Example of FMEA Table

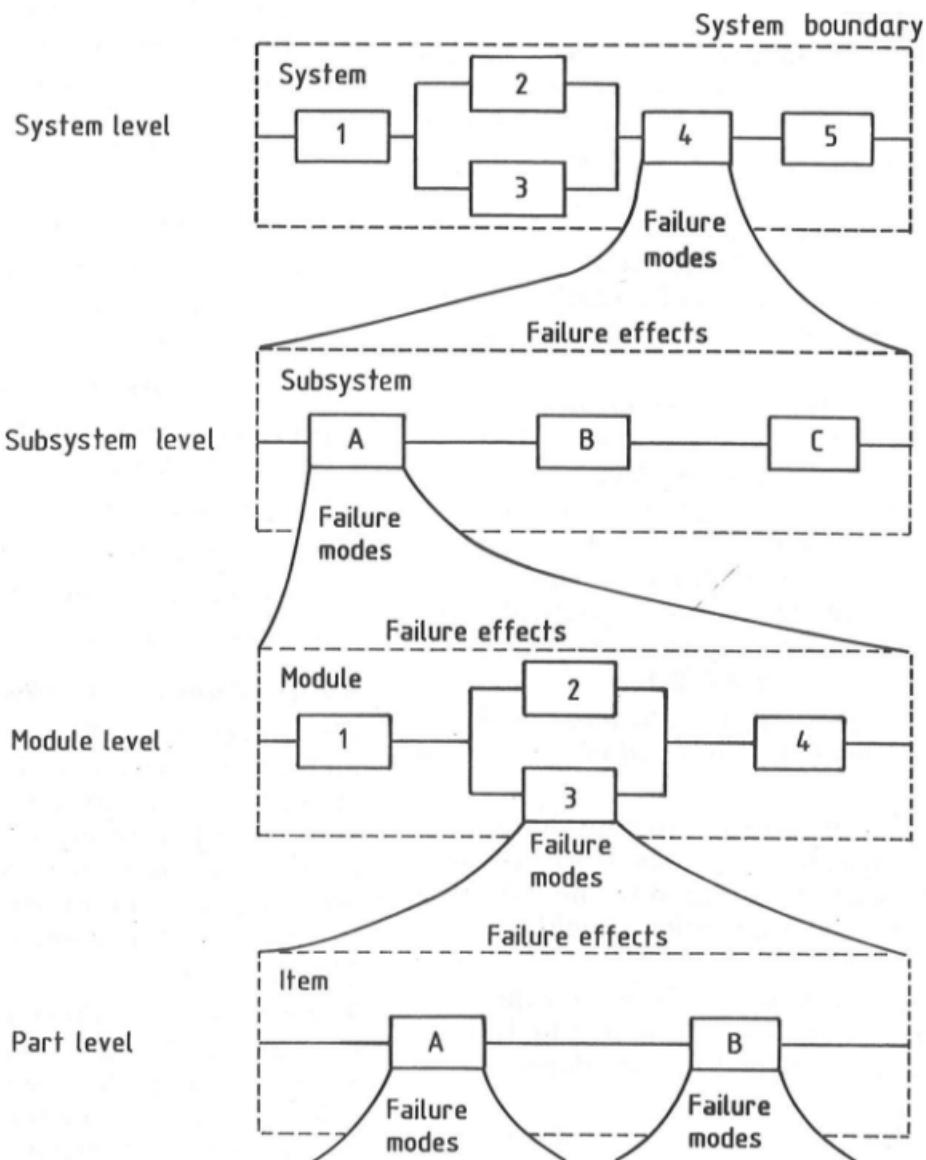


Figure 2.2: Criticality Grid

that define the structure of the system. Furthermore, the report should include a list of drawings (with issue status) on which the FMEA is based.

The summary section of the report should provide a brief explanation of the analysis method, the level to which it was carried out, the assumptions made, and the ground rules followed (see section 2.4.L). The summary should also include lists of the following:

1. Recommendations for designers, maintenance staff, planners, and users;
2. Failures that, when occurring alone, have significant consequences;
3. Failures that have no impact; and
4. Design changes that were made as a result of the FMEA (or FMECA).

## **2.4 Application**

### **2.4.1 Field of application**

FMEA is a method that is primarily adapted to the study of material and equipment failures and that can be applied to categories of systems based on different technologies (electrical, mechanical, hydraulic, etc.) and combinations of technologies. FMEA should also include the consideration of software and human performance where these are relevant to the reliability of the system. An FMEA can be a study for general use or it may be specific to particular pieces of equipment, to systems or to projects as a whole.

### **2.4.2 Application within a project**

The user should determine how and for what purposes he uses FMEA within his own technical discipline. It may be used alone or to complement and support other methods of reliability analysis. The requirements for FMEA originate from the need to understand hardware behavior and its implications for the operation of the system or equipment. The need for FMEA can vary widely from one project to another. FMEA is the principal reliability engineering activity in support of the design review concept (see 4.2.1.4 of BS 5760: Part 1: 1988) and should be put into use from the very first steps of system and subsystem design. FMEA is applicable to all levels of system design but is most appropriate for lower levels where large numbers of items are involved and/or there is functional complexity. Special training of personnel performing FMEA is essential and they need the close collaboration of systems engineers and designers. The

FMEA should be updated as the project progresses and as designs are modified. At the end of the project, FMEA is used to check the design and may be essential for demonstration of conformity of a designed system to the required standards, regulations, and user's requirements.

Information from the FMEA identifies priorities for statistical process control sampling and inspection tests during manufacture and installation and for qualification, approval, acceptance and start-up tests. It provides essential information for diagnostic and maintenance procedures for inclusion in handbooks.

In deciding on the extent and the way in which FMEA should be applied to an item or design, it is important to consider the specific purposes for which FMEA results are needed, the time phasing with other activities and the importance of establishing a predetermined degree of awareness and control over unwanted failure modes and effects. This leads to the planning of FMEA in qualitative terms at specified levels (system, subsystem, component, item) to relate to the iterative design and development process (see BS 5760: Part 1).

To ensure that it is effective, the place of FMEA should be clearly established in the reliability program, together with the time, manpower and other resources needed to make it effective. It is vital that FMEA is not abridged to save time and money. If time and money are short the FMEA should concentrate on those parts of the design which are new or are used in new ways. FMEA can be economically directed to areas identified as crucial by other methods of analysis, e.g. fault tree analysis (FTA).

### **2.4.3    Uses of FMEA**

Some of the detailed applications and benefits of FMEA are listed below:

- (a) to avoid costly modifications by the early identification of design deficiencies;
- (b) to identify failures which, when they occur alone or in combination, have unacceptable or significant effects, and to determine the failure modes which may seriously affect the expected or required operation;<sup>1</sup>
- (c) to determine the need for the following:
  - (1) redundancy;
  - (2) design improvement;
  - (3) more generous stress allowances (derating);
  - (4) screening of items;

<sup>1</sup> Such effects may include secondary failures.

- (5) design of features that ensure that the system fails in a preferred failure mode, e.g. 'fail-safe' outcomes of failures;
- (6) selection of alternative materials, parts, devices, and components;
- (d) to identify serious failure consequences and hence the need for changes in design and/or operational rules;
- (e) to provide the logic model required to evaluate the probability or rate of occurrence of anomalous operating conditions of the system in preparation for criticality analysis;
- (f) to disclose safety hazard, and product liability problem areas, or non-compliance with regulatory requirements; **Note:** Frequently, separate studies will be required for safety, but overlap is inevitable and therefore cooperation is highly advisable.
- (g) to ensure that the development test programme can detect potential failure modes;
- (h) to focus upon key areas in which to concentrate quality control, inspection and manufacturing process controls;
- (i) to assist in defining various aspects of the general maintenance strategy, such as:
  - (1) establishing the need for data recording and condition monitoring during testing, checking-out and use;
  - (2) provision of information for development of trouble-shooting guides;
  - (3) establishing maintenance cycles which anticipate and avoid wear-out failures;
  - (4) the selection of preventative or corrective maintenance schedules, facilities, equipment and staff;
  - (5) selection of built-in test equipment and suitable test points;
- (j) to provide a systematic and rigorous approach to the study of the installation in which the system is embedded;
- (k) to facilitate or support the determination of test criteria, test plans and diagnostic procedures, for example: performance testing, reliability testing;
- (l) to identify parts and assemblies requiring worst case analysis (frequently required for failure modes involving parameter drifts);

- (m) to support the design of fault isolation sequences and to support the planning for alternative modes of operation and reconfiguration;
- (n) to facilitate communication between the following:
  - (1) general and specialized engineers;
  - (2) equipment manufacturer and his suppliers;
  - (3) system user and the designer or manufacturer;
- (o) to enhance the analyst's knowledge and understanding of the behaviour of the equipment studied;
- (p) to provide designers with an understanding of the factors which influence the reliability of the system;
- (q) to provide a final document that is proof of the fact that (and of the extent to which) care has been taken to ensure that the design will meet its specification in service. (This is especially important in the case of product liability)

#### **2.4.4 Limitations and Drawbacks**

FMEA is extremely efficient when it is applied to the analysis of elements that cause a failure of the entire system or of a major function of the system. However, FMEA may be difficult and tedious for the case of complex systems that have multiple functions involving different sets of system components. This is because of the quantity of detailed system information that needs to be considered. This difficulty can be increased by the existence of a number of possible operating modes, as well as by consideration of the repair and maintenance policies. FMEA can be a laborious and inefficient process unless it is judiciously applied. The uses to which the results are to be put subsequently should be defined and FMEA should not be included in requirements specifications indiscriminately. Complications, misunderstandings and errors can occur when FMEA attempts to span several levels in a hierarchical structure if redundancy is applied in the system design. It is therefore preferable for an FMEA to be restricted to relating two levels only in the hierarchical structure. For example, it is a relatively straightforward task to identify failure modes of items and to determine their effects on the assembly. These effects then become the failure modes at the next level up, e.g. the module, and so on. However, successful multi-level FMEAs are often carried out. FMEA is applicable to all levels of a system but is most appropriate to lower levels where large numbers of items are involved and/or there is functional complexity.

#### **2.4.5 Relationships with Other Methods**

FMEA (or FMECA) can be used alone. As a systematic inductive method of analysis, FMEA is most often used to complement other approaches, especially deductive ones. At the design stage, it is often difficult to decide whether the inductive or deductive approach is dominant, as both are combined in processes of thought and analysis. Where levels of risk are identified in industrial facilities and systems, the inductive approach is preferred and therefore FMEA is an essential design tool. However, it should be supplemented by other methods. This is particularly the case when problems need to be identified and solutions need to be found in situations where multiple failures and sequential effects need to be studied. The method used first will depend on the project programme.

During the early design stages, where only functions, general system structure and subsystems have been defined, successful performance of the system can be depicted by a reliability block diagram or a failure path by a fault tree. However, to assist in drawing these diagrams

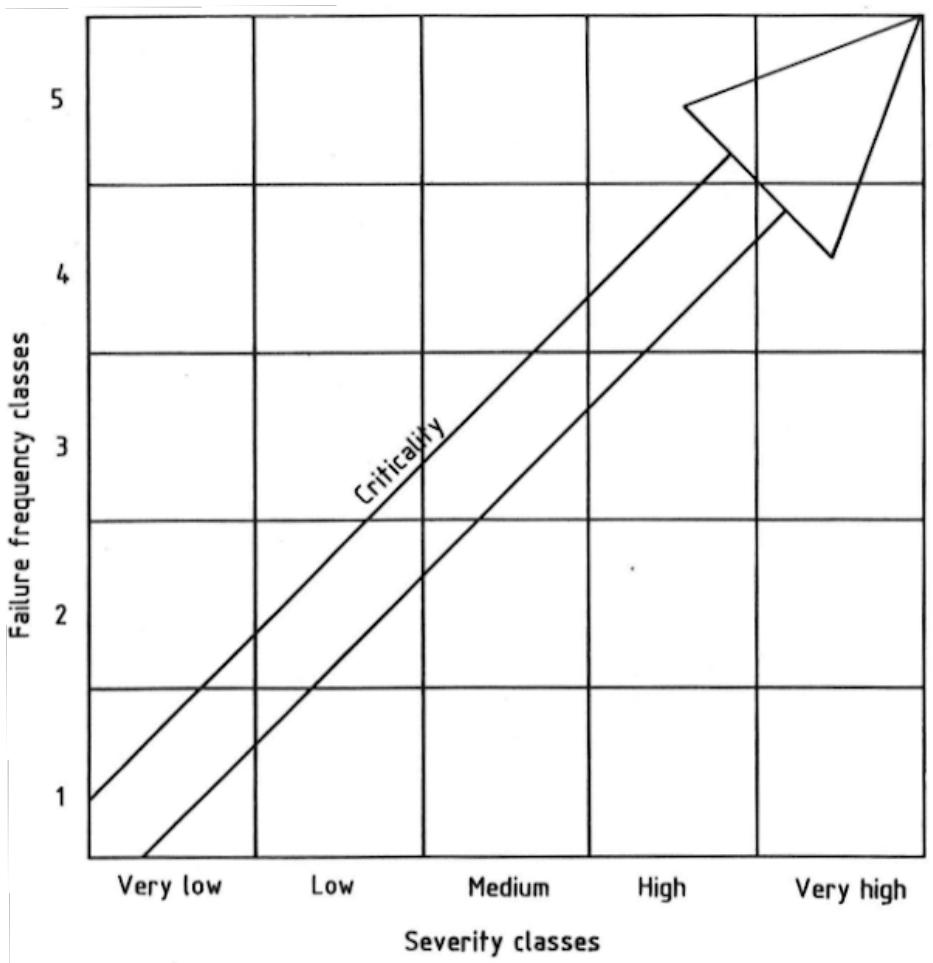


Figure 2.3: Criticality Grid

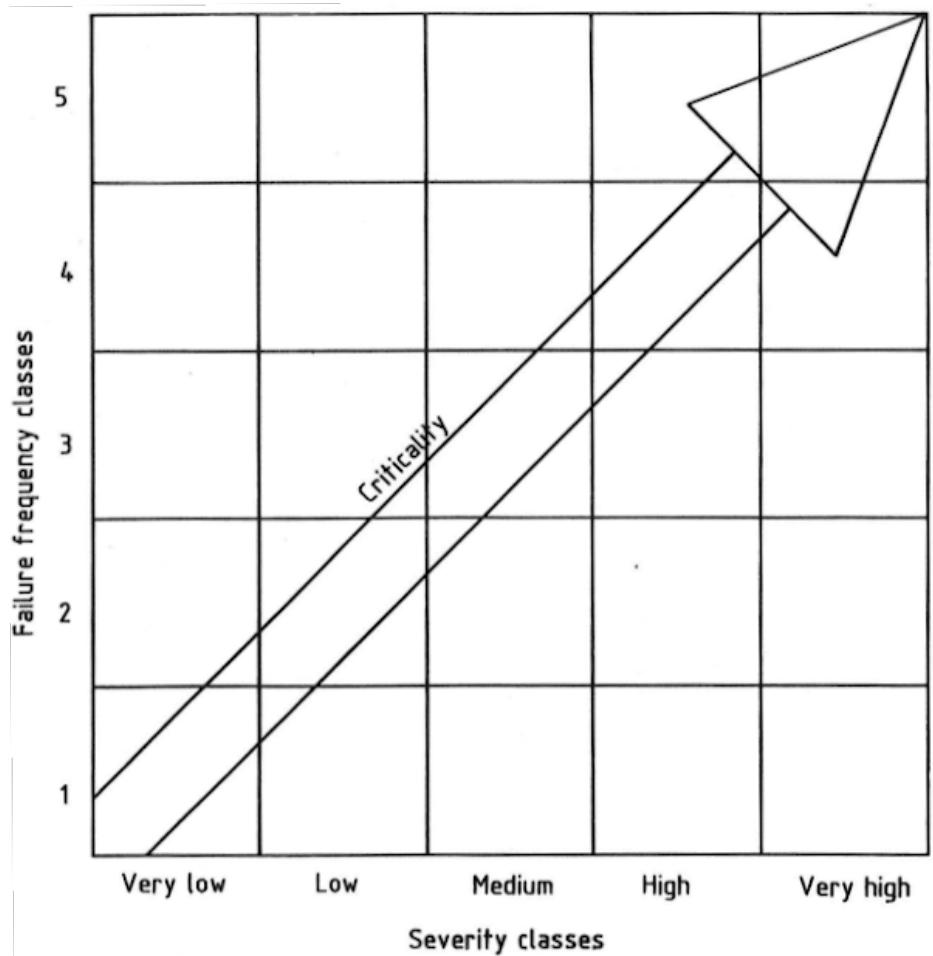


Figure 2.4: Criticality Grid

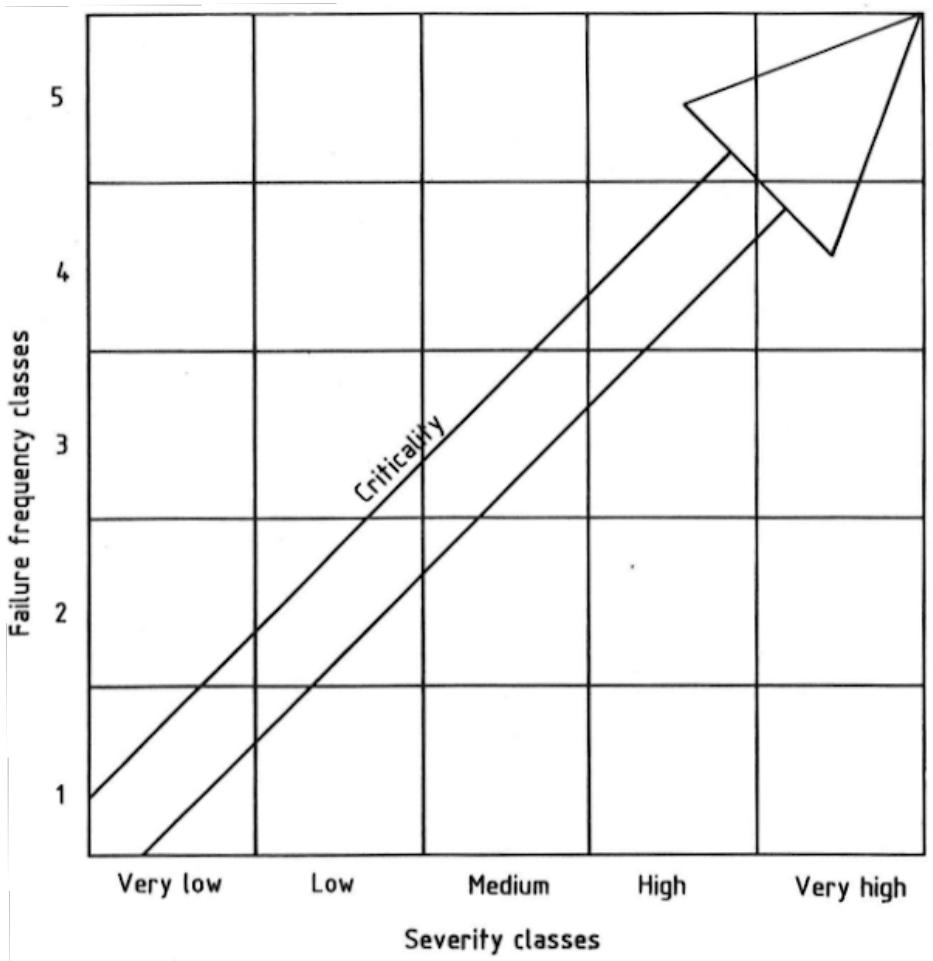


Figure 2.5: Criticality Grid

	1	2	3	4	5
Failure frequency classes	5	A	C	C	D
	4	A	B	C	C
	3	A	B	B	C
	2	A	A	B	B
	1	A	A	A	A
Severity classes					

Figure 2.6: FMEA Criticality Matrix

# CHAPTER 3

## HARDWARE CONTRIBUTION

### 3.1 Introduction

In the context of new applications for autonomous mobility, digital components are required to reach high levels of functional safety performances. This level of assurance is necessary to supply safely the computational power and advanced processing required by those applications. It is therefore necessary for digital SoCs safety engineers to be able to demonstrate thru advanced provable methods the achieved reliability of their system and counter measures.

Similarly to complex electomechanical systems, it is difficult to predict the failure modes of a complex SoC which exhibits an almost infinite state space (in the order of  $2^n$ ,  $n$  is the number of sequential elements, reaching ten's of thousands easily) and distributed-systems characteristics: numerous independents sub-systems operating and communicating asynchronously.

Digital systems are subject to two kind of errors, *permanent* which are created by destructive or aging effects, and *transient* [6] created by particle impacts such as thermal neutrons at ground level or solar wind in low and high earth orbits [71][51]. Permanent effects shows in the form of a permanent stuck to an electrical value ('0' or '1' logic value) and can occurs on any digital element (combinational logic, i.e. *logic gate* or sequential element, i.e. *flip-flops*). So do transient faults, also called *soft errors*, but with different, non-permanent, effects on logic or sequential elements. Transient faults on logic gates are called *Single Event Transient* (SET) [37] and are particularly dangerous on clock trees and reset trees (which distributes the clock and reset signals through the chip using trees of *buffers*) of SoCs as their effect, that has the form of a glitch, is to reset or desynchronize the sequencing of a sub-part of the system. Transient faults on sequential elements (memories or flip-flops) only invert the value of the element which will retain the faulty value until overwritten by a new value.

They are called *Single Event Upset* (SEU) and are the main cause of safety goals violations in digital SoCs [48].

However, digital system exhibit a natural resistance to soft errors and most of them have no functional effect while a small proportion of them ( $\approx 10\%$  of them in a standard 5-stages processor [62, 49]) will lead to system execution failure. FMEDA analysis [43], targeting goals such as ISO26262 automotive safety norm [35] certification will consist in quantifying those failure modes, proving the effectiveness of counter measures and absence of safety goals violation. An effective solution consists in submitting the system to faults, by simulation or under radiation beam, therefore stressing it and provoking intentionally dysfunctional behaviors. Those 'out of trajectory' behaviors can then be recorded, analysed and used for FMEDA analysis in the certification process. However, both methods are costly both in term of engineering setup needed and cost: fault injection of a full SoC requires a complex setup, test suite and costly hardware emulator while radiation test requires an acquisition system, a test setup and access to costly and constrained radiation facilities. Both have the disadvantage to require, the full SoC gate netlist (fault injection [67]) or silicon (irradiation [9]). Also, both methods can be classified as experimental as it is a verification 'by observation' of the resilience of the system to faults. No proof, except statistical confidence is made on the extracted faults metrics.

In this work, we aim to assess the capability of Model-Based Safety Assessment methods to build the dysfunctional model of a digital SoC from its subsystems and perform the currently hand-made FMEDA of the full system automatically. We expect the methods to be able to quantify globally the system safety metrics more accurately than with hand-made spreadsheets which only basically multiply probabilities. Automatic failure analysis such as fault trees extraction, fault sequences leading to unwanted events are also expected to be of great help during the certification process. The problem to solve is then to extract and build the required dysfunctional models of the different subsystems of the SoC and to properly expose the failure modes in the constructed models to be able to use existing model composition frameworks.

The document is organized as follow: we first present the system used as example and how fault injection is used to expose dysfunctional behaviors and extract a model. The chosen approach is then detailed reminding generic principles before explaining specific mechanisms put in place to model digital system. Finally, the document details fault injection campaign post-processing methods and obtained results. We compare composition results with fault injection performed on the full system used as a reference.

## 3.2 State-of-the-Art

### 3.2.1 Probabilistic Methods in Digital Systems Safety

Probabilistic methods [60] [61] have been developed to estimate propagation and masking rates of errors in gate netlists. Such approaches, restricted to combinational logic provide an helper to estimate certain metrics ( $\lambda_{spf}$ , i.e. *Dangerous Undetected* by a safety mechanism faults [36]) required in ISO26262, but are far from being able to provide metrics even at the sequential block level. Likewise, industrial formal proof tools [22] [69] are able to compute such metrics by using formal methods.

Methods like FIDES [44] [68] targets Commercial Off-the-Shelf (COTS) based Electronic Control Unit (ECU), with components failure rates extracted from available reliability databases. It takes into account systematic or aging failures but not transient effects such as soft-errors.

### 3.2.2 Formal Methods in Digital Systems Safety

Formal methods [12, 13] are mostly used on unitary blocks or functionalities to prove assertions (i.e. properties) expressed in linear [65] or branching [25] timing logic. When applied to safety, it comes to proving absence of safety goals violations that are expressed as assertions on outputs in the presence of faults. Tools like [22][69] are able to compute, given a nelist of logic gates and flip-flops and an initial state, the cone of influence of flip-flops or gates and whether a fault in such elements can propagate to a given output. Such structural analysis can perform *Out-of-Cone-of-Influence (COI)* fault analysis allowing to classify a fault as *safe* when it cannot reach a given output. Activation analysis determine whether a fault injected on a specific node can be activated. Propagability analysis determine if an activated fault in a COI can propagate to a strobed output and detection analysis determines if a fault will (always) be propagated and detected at the checker output. Such analysis can reveal what logic is covered by a safety mechanism or not. However, no formal methods is able to address such safety properties at SoC level.

### 3.2.3 Altarica

Functional safety objective is to identify the most probable failure combinations leading to a feared event. Model-Based Safety Analysis performs safety analysis by building dysfunctional models for each block of the considered system and using formal methods to combine and extract failure modes at the system level

[47]. MBSA introduces the use of high level modeling languages dedicated to functional safety analysis [53] [4] [11]. It allows extending classical methods such as FMEA or fault trees. These languages help capture system dynamics and how failures propagate inside it. Moreover, models support structural modeling allowing identifying and locating induced effects of a failure inside the architecture.

Altarica Dataflow (Fig. 3.1) is an event-driven asynchronous language that implements discrete variables with a finite number of values, leading to a finite number of combinations of state values and propagated flows, allowing theoretically to cover the entire system state space. AltaRica Dataflow is at the core of several Reliability, Availability, Maintainability and Safety (RAMS) environments: Cecilia OCAS (Dassault Aviation), SimfiaNeo (Airbus Protect), and Safety Designer (Dassault Systèmes)

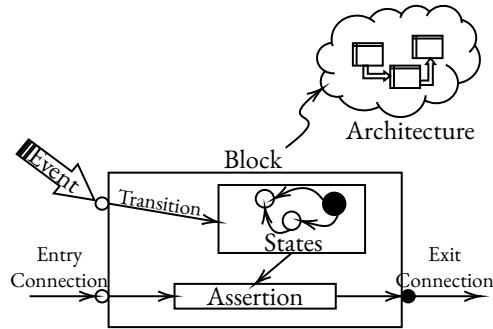


Figure 3.1: Altarica Dataflow Model

- *Variables:* AltaRica variables are discrete and represents an enumerated finite set of values called its *domain*. Variable definition inside its domain is free. The variable can represent for example functional modes, dysfunctional status, message types ....

Inside MBSA models, state variables are generally used to represent dysfunctional status with a default value as nominal behavior and a value for each degraded mode reached from any failure mode. Flow variables are generally used to describe the type of data exchanged between components. This type can represent a functional value (e.g. instruction value) or a dysfunctional value (e.g. message status). It depends of the model level of detail. As flows are used to propagate failures, they can be described either by sending a status or a faulty value.

- *Transitions:* Transitions describe possible states changing values. Transitions are guarded by a condition allowing the transition to become

fireable when true. A transition is associated with a triggering event and is fired when the event is triggered and the guard is true. In MBSA modeling, triggering events are used to represent failure modes. AltaRica allows to assign a probability law to an event, modeling the behavior of random failures or deterministic actions. The transition completion describes the effect of the failure mode on the component state. Guards can be enriched to restrict to describe conditional failures. For example, in a cold redundancy, some failures can't happen when the component is off.

- *Assertions*: Assertion is the mechanism used to set outputs values of a node. Output values are a function of input values and internal state values. Assertion can be interpreted as a logical function describing a truth table assigning outputs according to each combination of inputs and internal state values. Combinations are described through Boolean expressions and imperative programming constructs such as *if-then-else* or *case*.

Assertions are used to propagate failures from a faulty component to other blocks. Fault injected on the internal state is propagated to its output and then to others blocks. Depending of the granularity level of the model, assertions are manipulating either functional values or states.

### 3.3 First Manual Application of FMEA on a SoC

In order to fully understand the challenges that performing the full FMEA process implies (as depicted in chapter MISSIN REF), the decision of fully implement the process on a selected SoC has been taken. The SoC on choice has been the RISC-V based SoC developed in STMicroelectronics, in order to have full observability and have all the testing suite and reports of irradiation available. Figure 3.2 shows the complexity of the entire system of choice *SCR1*, composed of two cores and several peripherals, each of those dedicated to a specific communication or testing function.

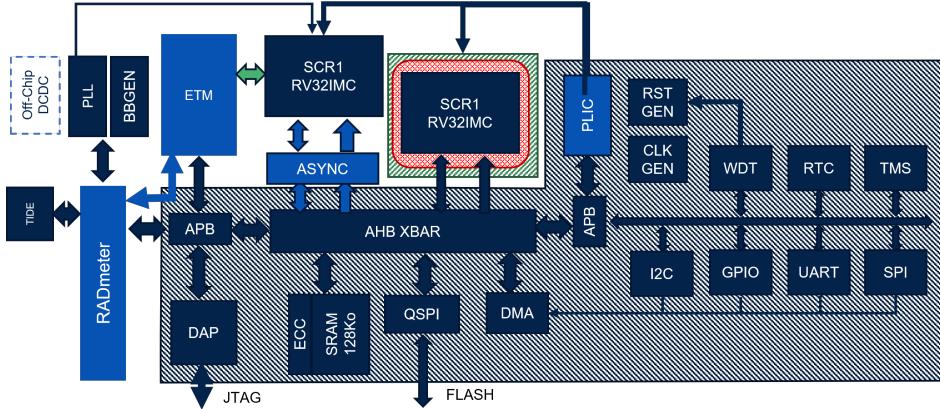


Figure 3.2: RTL Description of SCR1

It has been clear since the beginning that the greatest obstacle to overcome was the definition of the internal blocks of the cores, together with their interconnections. There is where the FMEA process has started. There was the need to identify one of the two twin cores and decompose it manually in order to obtain a complete definition to then analyse the possible failure modes. Figure 3.3 shows the beginning of the process, dictated by the identification of all the connections to peripherals that are established with the core of choice.

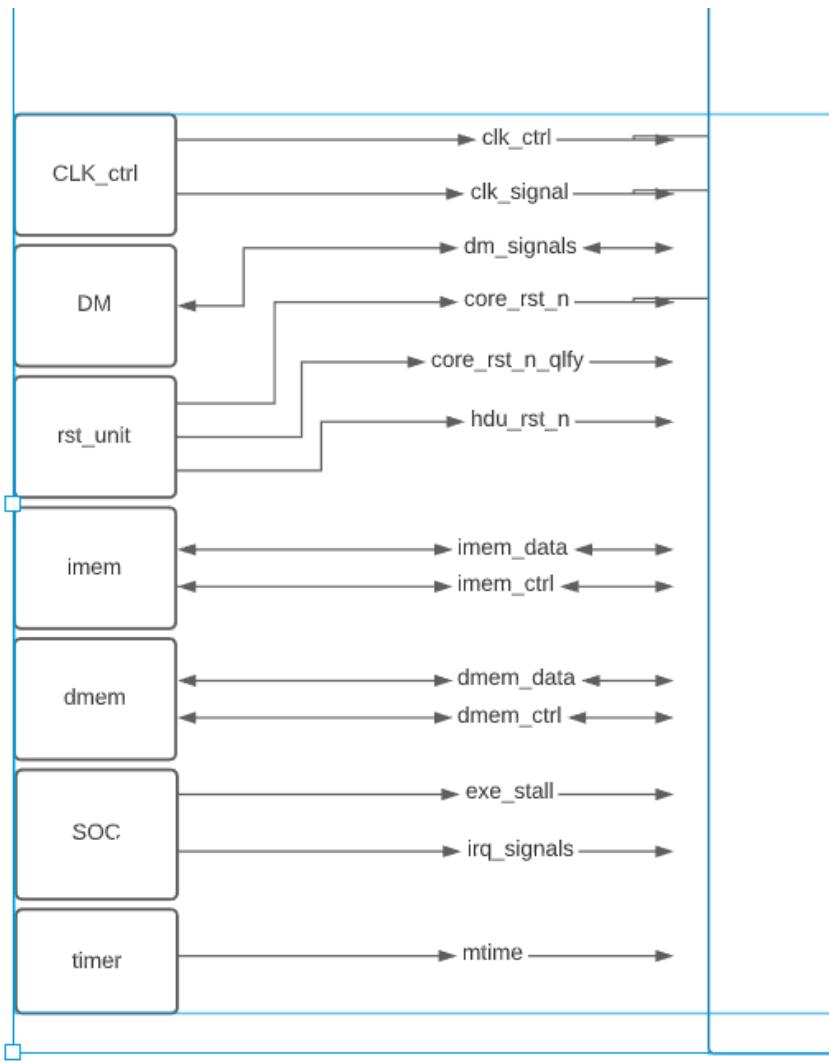


Figure 3.3: Inputs to the SoC

That being the starting point, the work has proceeded with the definition of all the internal signals to all the sub-blocks of which the core is composed. This is a rather time consuming operation to be carried out manually, and it will be shown, one of the most prone to introduce human errors. Nevertheless it has been completed and the results are shown in Fig. 3.4, showing how complex the interconnections could be even in case of a high level description of the internal of the core.

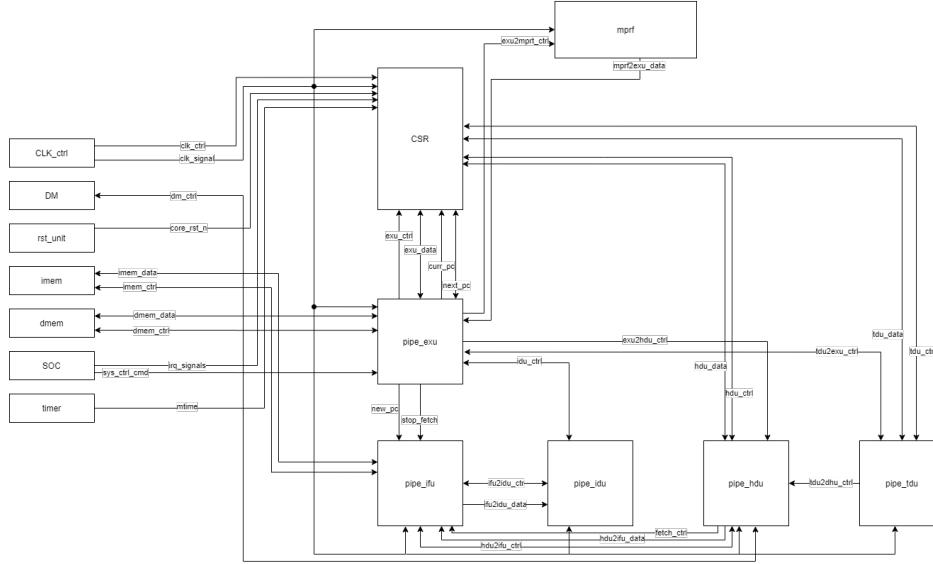


Figure 3.4: Detailed reconstruction of block model

Despite the completion of the internal signal definition, the complexity of the FMEA analysis procedure necessitated the use of the layout shown in Fig. 3.5 as a worksheet. This layout served as a guide for the FMEA analysis team, who were tasked with identifying potential failures within the system and assessing the severity of their impact. However, the complexity of the system, as evidenced by the intricate interconnections between sub-blocks, proved to be a significant obstacle for the team, and the procedure was ultimately halted due to its difficulty.

Systems2win		Process FMEA											
Title <Header Title>		Sample Product or Process - Rev C											
Program <Value Stream, Program, Product Family, Customer >													
Part Number <Part Number - perhaps with Engineering Change Revision Number and/or Date>													
Level / Phase	Choose from Set	Key Date <date>	Revision Number	<Revision #>	Document Number	<Document Control Number>							
Controlled? <NA (not controlled)		Expiration Date <date>	Revised <date>	Userd <User default date>	Core Team	<Core Team member names>							
Author <name>		Responsible <name>	Date (orig)										
2	Step 3 - Function Analysis	Requirement ID	Requirement From Lower Level	Potential Failure Analysis	Potential Failure Mode	Potential Failure Effect	Severity	Probability	Prevention Method	Detection Method	Actions Recommended	Who	When
Period	Step	Function To Parent Level	Requirement ID	Requirement From Lower Level	Potential Failure Mode	Potential Failure Effect	Severity	Probability	Prevention Method	Detection Method	Actions Recommended	Who	When
Door Finish Line for a U.S. Standard, Architectural Type, 64" H by 42" W by 2.5" T													
Process Step # 140 of door finish line													
Cut 4, equal height, 5x5", Anch, door hinge	Meet U.S. standard, hinge code requirements	14	Top hinge placement (see Figure 1)	Does not meet top of hinge to top of frame dimension	Fails to meet U.S. standard, hinge code requirements	9	9	incorrect requirement code template selected	2 NC program inputs modified by sensor input (D-C0010Cncl)	Machine lock-out if clutch is detected by sensor input (D-C0010Cncl)	1 No further action needed	No	On-going
		15	Bottom hinge placement (see Figure 1)	Does not meet bottom of hinge to finished floor dimension	Door does not close properly	7	7	incorrect requirement code template selected	2 NC program inputs modified by sensor input (D-C0010Cncl)	Scan traverse for dimensions or US Code requirement	8 No	No	On-going
					Door not sealed to template	5	5	5 SMT Procedure D-C0034Cba	Machine lock-out if clutch is detected by sensor input (D-C0010Cncl)	Visual inspection	2 No	No	On-going
						6	6	6	6	1	1	4/2/2010	
						7	7	7	7	2	2	4/2/2010	
						8	8	8	8	3	3	4/2/2010	
						9	9	9	9	4	4	4/2/2010	
						10	10	10	10	5	5	4/2/2010	
						11	11	11	11	6	6	4/2/2010	
						12	12	12	12	7	7	4/2/2010	
						13	13	13	13	8	8	4/2/2010	
						14	14	14	14	9	9	4/2/2010	
						15	15	15	15	10	10	4/2/2010	
						16	16	16	16	11	11	4/2/2010	
						17	17	17	17	12	12	4/2/2010	
						18	18	18	18	13	13	4/2/2010	
						19	19	19	19	14	14	4/2/2010	
						20	20	20	20	15	15	4/2/2010	
						21	21	21	21	16	16	4/2/2010	
						22	22	22	22	17	17	4/2/2010	
						23	23	23	23	18	18	4/2/2010	
						24	24	24	24	19	19	4/2/2010	
						25	25	25	25	20	20	4/2/2010	
						26	26	26	26	21	21	4/2/2010	
						27	27	27	27	22	22	4/2/2010	
						28	28	28	28	23	23	4/2/2010	
						29	29	29	29	24	24	4/2/2010	
						30	30	30	30	25	25	4/2/2010	
						31	31	31	31	26	26	4/2/2010	
						32	32	32	32	27	27	4/2/2010	
						33	33	33	33	28	28	4/2/2010	
						34	34	34	34	29	29	4/2/2010	
						35	35	35	35	30	30	4/2/2010	
						36	36	36	36	31	31	4/2/2010	
						37	37	37	37	32	32	4/2/2010	
						38	38	38	38	33	33	4/2/2010	
						39	39	39	39	34	34	4/2/2010	
						40	40	40	40	35	35	4/2/2010	
						41	41	41	41	36	36	4/2/2010	
						42	42	42	42	37	37	4/2/2010	
						43	43	43	43	38	38	4/2/2010	
						44	44	44	44	39	39	4/2/2010	
						45	45	45	45	40	40	4/2/2010	
						46	46	46	46	41	41	4/2/2010	
						47	47	47	47	42	42	4/2/2010	
						48	48	48	48	43	43	4/2/2010	
						49	49	49	49	44	44	4/2/2010	
						50	50	50	50	45	45	4/2/2010	
						51	51	51	51	46	46	4/2/2010	
						52	52	52	52	47	47	4/2/2010	
						53	53	53	53	48	48	4/2/2010	
						54	54	54	54	49	49	4/2/2010	
						55	55	55	55	50	50	4/2/2010	
						56	56	56	56	51	51	4/2/2010	
						57	57	57	57	52	52	4/2/2010	
						58	58	58	58	53	53	4/2/2010	
						59	59	59	59	54	54	4/2/2010	
						60	60	60	60	55	55	4/2/2010	
						61	61	61	61	56	56	4/2/2010	
						62	62	62	62	57	57	4/2/2010	
						63	63	63	63	58	58	4/2/2010	
						64	64	64	64	59	59	4/2/2010	
						65	65	65	65	60	60	4/2/2010	
						66	66	66	66	61	61	4/2/2010	
						67	67	67	67	62	62	4/2/2010	
						68	68	68	68	63	63	4/2/2010	
						69	69	69	69	64	64	4/2/2010	
						70	70	70	70	65	65	4/2/2010	
						71	71	71	71	66	66	4/2/2010	
						72	72	72	72	67	67	4/2/2010	
						73	73	73	73	68	68	4/2/2010	
						74	74	74	74	69	69	4/2/2010	
						75	75	75	75	70	70	4/2/2010	
						76	76	76	76	71	71	4/2/2010	
						77	77	77	77	72	72	4/2/2010	
						78	78	78	78	73	73	4/2/2010	
						79	79	79	79	74	74	4/2/2010	
						80	80	80	80	75	75	4/2/2010	
						81	81	81	81	76	76	4/2/2010	
						82	82	82	82	77	77	4/2/2010	
						83	83	83	83	78	78	4/2/2010	
						84	84	84	84	79	79	4/2/2010	
						85	85	85	85	80	80	4/2/2010	
						86	86	86	86	81	81	4/2/2010	
						87	87	87	87	82	82	4/2/2010	
						88	88	88	88	83	83	4/2/2010	
						89	89	89	89	84	84	4/2/2010	
						90	90	90	90	85	85	4/2/2010	
						91	91	91	91	86	86	4/2/2010	
						92	92	92	92	87	87	4/2/2010	
						93	93	93	93	88	88	4/2/2010	
						94	94	94	94	89	89	4/2/2010	
						95	95	95	95	90	90	4/2/2010	
						96	96	96	96	91	91	4/2/2010	
						97	97	97	97	92	92	4/2/2010	
						98	98	98	98	93	93	4/2/2010	
						99	99	99	99	94	94	4/2/2010	
						100	100	100	100	95	95	4/2/2010	

Figure 3.5: Classic FMEA Worksheet layout

Even with the aid of the layout, the team found that the manual identification of potential failures was too error-prone and time-consuming, necessitating

a re-evaluation of the FMEA analysis process. A first detailed analysis of the possible failure modes and their gravity has been completed, and one extract can be seen in the Fig. 3.6 Despite this setback, the layout shown in Fig. 3.4 remains a valuable tool for visualizing the internal workings of the core and can aid in future system design and analysis efforts.

PROBE	NOTES	NOTES	Failure Mode
<b>DATA Memory</b>			
1. tb_top.URV250ULL.SER1.dmem_burst	is the address bus, which size is defined by size	DATA 3 bits?	wrong data length misaligned/memroy read
2. tb_top.URV250ULL.SER1.dmem_burst	indicates the transaction burst size		wrong burst length errors
3. tb_top.URV250ULL.SER1.dmem_imatchlock	locked memory transfer sequence		?
4. tb_top.URV250ULL.SER1.dmem_imatchlock	provides protection information [10..user or handle]		fetching a wrong size/seed data
5. tb_top.URV250ULL.SER1.dmem_imatchlock	size of the transfer		faulty data reading
6. tb_top.URV250ULL.SER1.dmem_imatchlock	current transfer type		Silent data corruption
7. tb_top.URV250ULL.SER1.dmem_imatchlock	data bus type		Wrong reading for data
8. tb_top.URV250ULL.SER1.dmem_imatchlock	bus width		Wrong alignment
9. tb_top.URV250ULL.SER1.dmem_imatchlock	bus width		Wrong alignment
10. tb_top.URV250ULL.SER1.dmem_imatchlock	bus width		Wrong alignment
11. tb_top.URV250ULL.SER1.dmem_imatchlock	bus width		Wrong alignment
12. tb_top.URV250ULL.SER1.dmem_imatchlock	bus width		Wrong alignment
13. tb_top.URV250ULL.SER1.dmem_imatchlock	bus width		Wrong alignment
14. tb_top.URV250ULL.SER1.dmem_imatchlock	bus width		Wrong alignment
15. tb_top.URV250ULL.SER1.dmem_imatchlock	bus width		Wrong alignment
16. tb_top.URV250ULL.SER1.dmem_imatchlock	bus width		Wrong alignment
17. tb_top.URV250ULL.SER1.dmem_imatchlock	bus width		Wrong alignment
18. tb_top.URV250ULL.SER1.dmem_imatchlock	bus width		Wrong alignment
19. tb_top.URV250ULL.SER1.dmem_imatchlock	bus width		Wrong alignment
20. tb_top.URV250ULL.SER1.dmem_imatchlock	bus width		Wrong alignment
21. tb_top.URV250ULL.SER1.dmem_imatchlock	bus width		Wrong alignment
22. tb_top.URV250ULL.SER1.dmem_imatchlock	bus width		Wrong alignment
23. tb_top.URV250ULL.SER1.dmem_imatchlock	bus width		Wrong alignment
<b>INSTRUCTION Memory</b>			
1. tb_top.URV250ULL.SER1imem_burst	is the address bus, which size is defined by size	INSTRUCTION ? code dependent?	wrong instruction length wrong burst length errors
2. tb_top.URV250ULL.SER1imem_burst	indicates the transaction burst size		?
3. tb_top.URV250ULL.SER1imem_burst	locked memory transfer sequence		fetching a wrong size/seed data
4. tb_top.URV250ULL.SER1imem_burst	provides protection information [10..user or handle]		faulty data reading
5. tb_top.URV250ULL.SER1imem_burst	size of the transfer		data corruption
6. tb_top.URV250ULL.SER1imem_burst	current transfer type		data corruption
7. tb_top.URV250ULL.SER1imem_burst	microarchitectural register (Core in the design)		?
8. tb_top.URV250ULL.SER1imem_burst	microarchitectural register (Core in the design)		?
9. tb_top.URV250ULL.SER1imem_burst	microarchitectural register (Core in the design)		?
10. tb_top.URV250ULL.SER1imem_burst	microarchitectural register (Core in the design)		?
11. tb_top.URV250ULL.SER1imem_burst	microarchitectural register (Core in the design)		?
12. tb_top.URV250ULL.SER1imem_burst	microarchitectural register (Core in the design)		?
13. tb_top.URV250ULL.SER1imem_burst	microarchitectural register (Core in the design)		?
14. tb_top.URV250ULL.SER1imem_burst	microarchitectural register (Core in the design)		?
15. tb_top.URV250ULL.SER1imem_burst	microarchitectural register (Core in the design)		?
16. tb_top.URV250ULL.SER1imem_burst	microarchitectural register (Core in the design)		?
17. tb_top.URV250ULL.SER1imem_burst	microarchitectural register (Core in the design)		?
18. tb_top.URV250ULL.SER1imem_burst	microarchitectural register (Core in the design)		?
19. tb_top.URV250ULL.SER1imem_burst	microarchitectural register (Core in the design)		?
20. tb_top.URV250ULL.SER1imem_burst	microarchitectural register (Core in the design)		?
21. tb_top.URV250ULL.SER1imem_burst	microarchitectural register (Core in the design)		?
22. tb_top.URV250ULL.SER1imem_burst	microarchitectural register (Core in the design)		?
23. tb_top.URV250ULL.SER1imem_burst	microarchitectural register (Core in the design)		?
<b>CSR</b>			
1. tb_top.URV250ULL.SER1.core_csr_base	status register of the core	CSR search register (CP)	warning
2. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt enable		?
3. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt disable		?
4. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt enable		?
5. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt disable		?
6. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt enable		?
7. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt disable		?
8. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt enable		?
9. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt disable		?
10. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt enable		?
11. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt disable		?
12. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt enable		?
13. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt disable		?
14. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt enable		?
15. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt disable		?
16. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt enable		?
17. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt disable		?
18. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt enable		?
19. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt disable		?
20. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt enable		?
21. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt disable		?
22. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt enable		?
23. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt disable		?
<b>Machine interrupt</b>			
1. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt enable		warning
2. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt disable		?
3. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt enable		?
4. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt disable		?
5. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt enable		?
6. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt disable		?
7. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt enable		?
8. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt disable		?
9. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt enable		?
10. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt disable		?
11. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt enable		?
12. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt disable		?
13. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt enable		?
14. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt disable		?
15. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt enable		?
16. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt disable		?
17. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt enable		?
18. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt disable		?
19. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt enable		?
20. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt disable		?
21. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt enable		?
22. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt disable		?
23. tb_top.URV250ULL.SER1.core_csr_base	machine interrupt enable		?

Figure 3.6: Classic FMEA Worksheet of SCR1

The difficulty encountered during the FMEA analysis procedure highlights the need to re-think the overall approach to system design and analysis. While the layout shown in Fig. 3.4 provides a useful tool for visualizing system in-

terconnections, it also underscores the need for more automated and efficient methods of identifying potential failures. By relying on manual analysis, the FMEA procedure is susceptible to human error and can be prohibitively time-consuming for complex systems like the one depicted in the figure. Moving forward, it may be necessary to explore more sophisticated analysis methods, such as automated approaches, to ensure that potential failures are identified and addressed in a timely and accurate manner.

### 3.4 Modeling Digital Systems for MBSA

Digital systems, by essence, lend themselves well to finite state machines representation making the use of languages and formalism such as Altarica very suitable for their modelling. However, dysfunctional modeling requires extracting the faulty behaviour of the blocks composing the system. Such task is usually carried out by a Failure Mode and Effect Analysis (FMEA) to identify possible malfunction of the individual blocks. In digital system, such task can be performed automatically by simulation with fault injection[38] and possibly formal methods [54].

The main issue in modelling digital systems for MBSA is choosing the adequate level of abstraction avoiding a direct  $1 \Leftrightarrow 1$  translation of *Hardware Description Languages* (HDL) modeling concepts into Altarica. When extracting a safety model from a digital block three points must be addressed:

- Structural hierarchy: Because Altarica support hierarchy [59], translating hierarchy with adequate granularity can be straightforward, especially as natural design hierarchy is usually a good candidate.
- Behavioral modelling: Faulty behavioral aspects requires extraction of failure modes which can be performed manually, based on design knowledge or automatically using fault injection or formal approaches. Fault injection is well suited to such analysis, especially in the world of digital design which rely heavily on HDL simulators and digital fault injection driven by ISO26262 requirements. In this work we will exclusively focus on fault injection.
- Faults propagation: Blocks in a SoCs are usually connected though buses with well defined protocols and their failures modes (*unaligned access ...*) are known. The issue comes in the granularity of the modelling that, if too low will lead to too numerous events (1 HDL signal  $\rightarrow$  1 flow variable) while a too high abstraction may prevent catching of some protocol failures.

Fault injection campaigns are used to characterize the behavior of the system from its output pins point of view which are the 'vectors' for faults propagation between blocks. Also, knowing the functionality of each of these pins, it is possible to attach some possible consequences to the failure to such (group of) output(s). Such semantic labelling is, however, still manual and based on safety engineers knowledge and experience.

### 3.5 Methodology

On top of any explicit finite state machine or control code encoding the user specified behaviour, a more complete state exist that includes the totality of the signals belonging to the control path of a design, such as data states implicitly exposed in controls states, or implicitly coded control states. These signals compose a more complete and larger state machine exposing new states and transitions that are implicitly specified, for example resulting from Cartesian product of automatons. Those are, technically, the signals driving transitions conditions.

Combinations of these signals in those states can lead to a subtle set of fault states, difficult to identify from the HDL description as the encoding in this state machine is sparse due to correlations. Such argument comes from the fact that even for a small ( $>\approx 50$ ) number of flip-flops, the complete state space ( $2^{50}$ ) cannot be traversed in a reasonable time. Therefore a non-negligible proportion of these states are what we call *illegal states* i.e. unreachable under normal behaviour, potentially leading to undesired and unspecified behavior when the block is exposed to those states though faults.

In order to build a failure model from a nominal behavioral *Register Transfer Model* (RTL) in Verilog or VHDL, behaviour of the system under faults must be analyzed and faulty behavior as well as failure modes must be extracted. We proceed using the following steps:

1. *Identification and Extraction of State Signals*: Starting from the functional description, the set of flip-flops, belonging to both the *control* and possibly *data* paths composing what we name as the *state*, has to be identified and extracted. This set, composed by all the flip-flops composing the control path and possibly the datapath which maintain the control state of the block, correspond to possible fault injection sites.
2. *Testbench Setup* : A standalone testbench is set up with care given to coverage and testbench representability as the states traversed during this golden execution will serve as non-faulty reference behavior. Tools like

*Incisive Metrics Coverage* (IMC) [14] or *Certitude* [57] can be used to assess testbench coverage. A first reference run is performed to allow extraction of golden functional states that will be used later in the process to be differentiated from non-functional ones under fault injection.

3. *Fault Injection Campaign*: Fault injection is the mean by which the misbehavior and faulty execution is exposed on purposes. Probes (i.e., observation points) are defined during the setup of the fault injection campaign. They are set on the outputs of all blocks in order to identify failures that propagates to other blocks. Probes monitor and compare the probed signal value at each clock cycle with the golden reference and report any difference. They have been set to stop simulation when a fault reaches an output of the design. This step is the core of our analysis aimed at extracting faulty behaviour, modes and effects though exploration of the faulty states by fault injection.
4. *Extraction of Faulty Behavior*: Once the faulty runs have completed, non-functional (i.e. *faulty*) states and behavior are extracted by subtracting functional (*golden*) states taken from the golden run state dictionary to the faulty run states, leaving only newly discovered faulty states and transitions.
5. *Construction of the Faulty Model*: The newly discovered states and transitions are used to augment the functional models with faulty behavior. Transitions from a functional to a non-functional state are labeled with the responsible faults so are states responsible for an incorrect output. This model serves as a base for the translation into the Altarica language.

Currently, the method is limited in the *effect* analysis of the FMEA. Effect such as *loss of power* cannot be attached automatically to a faulty state as it would requires an inference and abstraction process out the reach of the tool currently. Thus, such labelling is performed manually by attaching effects to outputs and then back-propagating them into the states and faults responsible for the given outputs corruptions.

### 3.5.1 Faulty Behavior Model Construction

Once the faulty behavior has been extracted from faulty runs, the faulty model can be constructed using graph analysis algorithms. The first step in the model construction is collapsing states that are not meaningful for the dysfunctional model. We proceed currently with the following rules:

- Any component (connected subgraph) comprising only legal states and legal transitions are collapsed into one single *functional* state.
- Legal states with illegal transitions or incorrect outputs (outputs values do differs from reference in these states) are kept and illegal transition probabilities are attached.
- Any component comprising only nodes not propagating any faults to outputs are collapsed into one single *faulty* state. Probabilities to enter this state can be extracted from transitions leading to the collapsed states.
- Faulty nodes propagating faults to outputs are kept and transitions probabilities are attached to allow computing incorrect outputs probabilities.
- Effect attached to output pins are back propagated in the state graph faulty states where output corruptions occurs.

However additional rules may be added like to remove faulty nodes and transitions from masked faults for example, especially those not leading to any latent faults (execution is correct with no faults propagated to outputs and internal state doesn't differs from reference one at some point, i.e. fault has *vanished*). We ultimately target discrete-time Markov chains [18] for our dysfunctional behaviour modelling (Fig. 3.7).

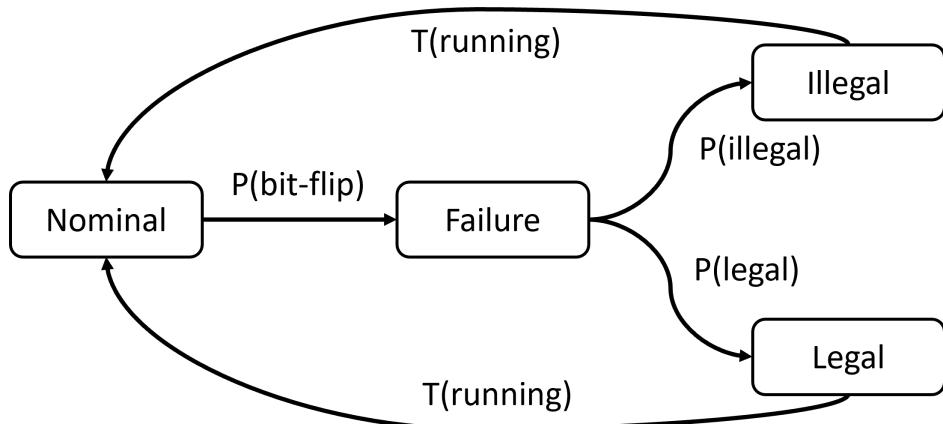


Figure 3.7: Altarica Dysfunctional Model

### 3.5.2 Completeness of Extraction

The main risk in state identification is to under or overestimate the state which would lead to uncovered faulty states (fault not injected in a flip-flop misidentified as not *control*) or over estimate the state leading to classification of what

are, in fact, data state as control states. The latter can be easily identified as randomizing data in the golden or faulty state machine extraction step leads to an increasing number of states with the number of runs. On Fig. 3.8, a correct identification leads to a saturating number of states (green curves) while an incorrect one leads to a diverging number of states as the number of tests grows (red curve).

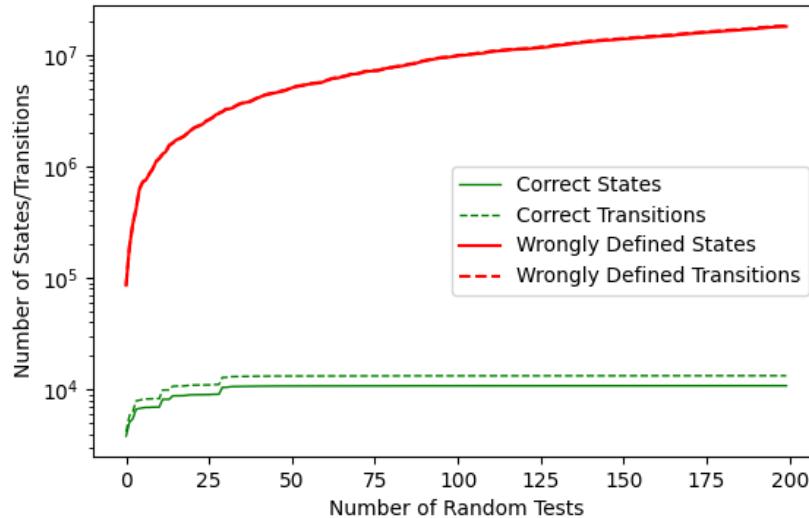


Figure 3.8: Completeness of the Extraction

### 3.5.3 Altarica Modeling

Such an automaton representation is adequate for Altarica modelling as described below. When performing translation to Altarica, two elements shall be extracted:

1. The internal state machine corresponding to failures.
2. The assertion part corresponding to the propagation failure probability from input to one or more output of element.

Base modelling must include at least four states (Fig. 3.15): a *nominal* state where no failure occurs, a *failure* state corresponding to a bit-flip error injection and an *illegal* state corresponding to propagation of the failure to one or more outputs. The *legal* state correspond to failures leading to legal transitions, without failure propagation to outputs.

Assertions on outputs are conditioned by the internal state machine and inputs of the block. Every time internal state machine is in the illegal state,

outputs values are updated. In same way, if one input of the block is set in the failed state, outputs are updated. Probabilities to generate a faulty output or to propagate failures from inputs to outputs are extracted from fault injection campaigns (Table ??). Currently, all illegal states are collapsed into a single one, but different non-functional states corresponding to different failure modes can be extracted as well, such as represented on Fig. 3.15 where two illegal states are identified whether or not a simulation timeout (10% of golden execution time) occurs. Criteria for refined dysfunctional automaton extraction are not yet addressed as well as construction and reduction rules from fault injection data for such an automaton.

## 3.6 Application: I<sub>2</sub>C to AHB bridge

In order to exercise the methodology presented in Section 4.3, we use a test case composed of 2 blocks: an *I<sub>2</sub>C* slave [42] connected to an *AHB* [29] bus master interface (Fig. 3.9). Commands (*read* or *write*) along with parameters (*address* and *data*) are received on the serial line and transformed into a series of *AHB* read and write transactions. Such a system, composed of two interconnected blocks, is humanely understandable so are its dysfunctional modes, while being complex enough to detail thoroughly the methodology.

The *I<sub>2</sub>C* slave, taken from [34], receives *read* or *write* commands followed by an address byte and an optional data byte. On an *I<sub>2</sub>C* read, the byte returned from the *AHB* read transaction is returned. chronogramm for the read and write sequences are represented on Fig. 3.10. The system is represented on Fig. 3.9. At both end of the system (*I<sub>2</sub>C* input and *AHB* output buses), *I<sub>2</sub>C* master and *AHB* slave Verification IPs (VIP) are attached to generate and verify correctness of *I<sub>2</sub>C* and *AHB* transactions.

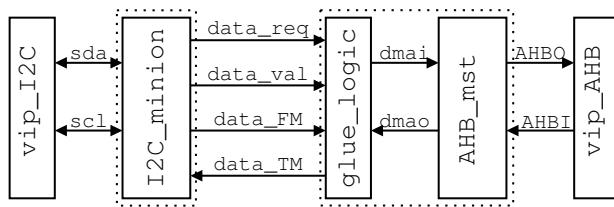


Figure 3.9: I<sub>2</sub>C to AHB System Block Diagram

### 3.6.1 I<sub>2</sub>C Block Modeling

The testbench is composed of a series of read and write random transactions. The coverage evaluation of the design has been carried out, results are presented

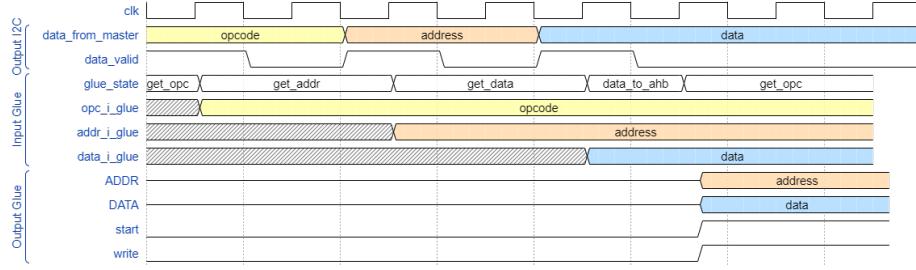


Figure 3.10: I<sub>2</sub>C/AHB System Model

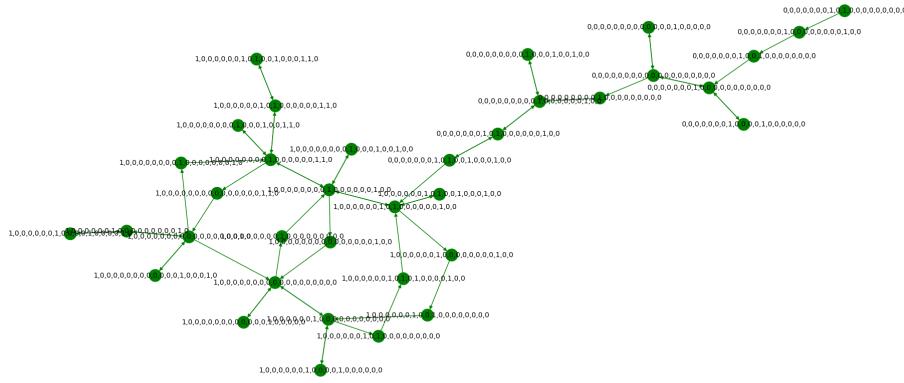


Figure 3.11: I<sub>2</sub>C Gold Automaton

in Table 3.1. Having considered the results of the coverage evaluations sufficient for the demonstration, application of the method presented in section 4.3 have been performed. The list of all injection sites, reported by Cadence *Xcelium Fault Simulator* (FSV) [14] fault injection tool are considered for state including ones containing data as the serial nature of the I<sub>2</sub>C protocol, which mixes control and data frames on the same signals thought time-multiplexing, doesn't allow differentiation. However, the small size of data considered (8-bit) only induce a low (256) superset of the real control states. All outputs are probed so that any mismatch with the reference run will stop the simulation and report the fault as *Detected*. State (flip-flop value, i.e. '0' or '1') is simply extracted at each clock cycle and printed in the simulation logs to be post-processed.

Fault injection traces are then processed following rules described in section 3.5.1 extracting transitions probabilities between the connected subgraphs:

- Nominal 1 - Subgraph made of legal states only, part of the nominal execution.

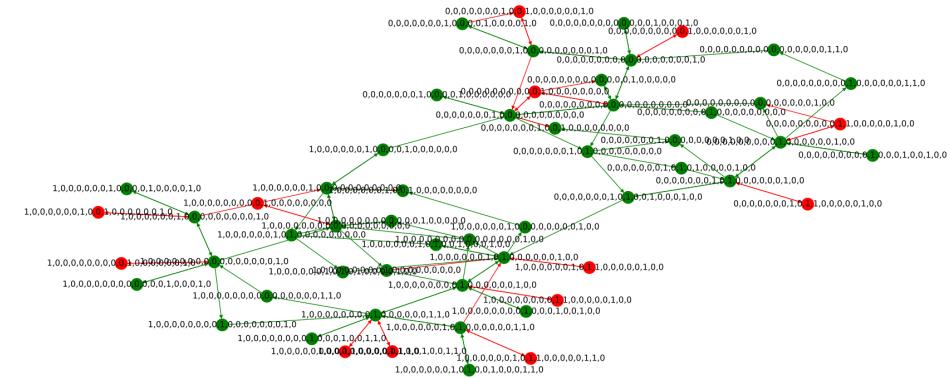


Figure 3.12: I<sub>2</sub>C Complete Automaton

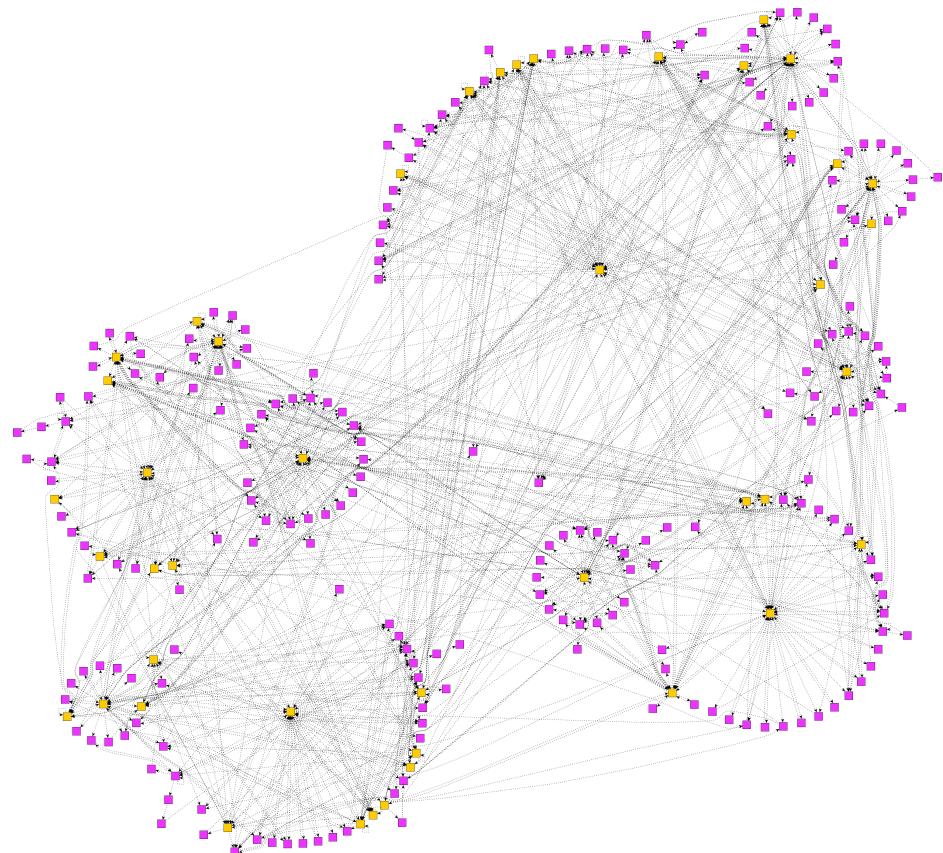


Figure 3.13: I<sub>2</sub>C Complete Automaton Clusterized

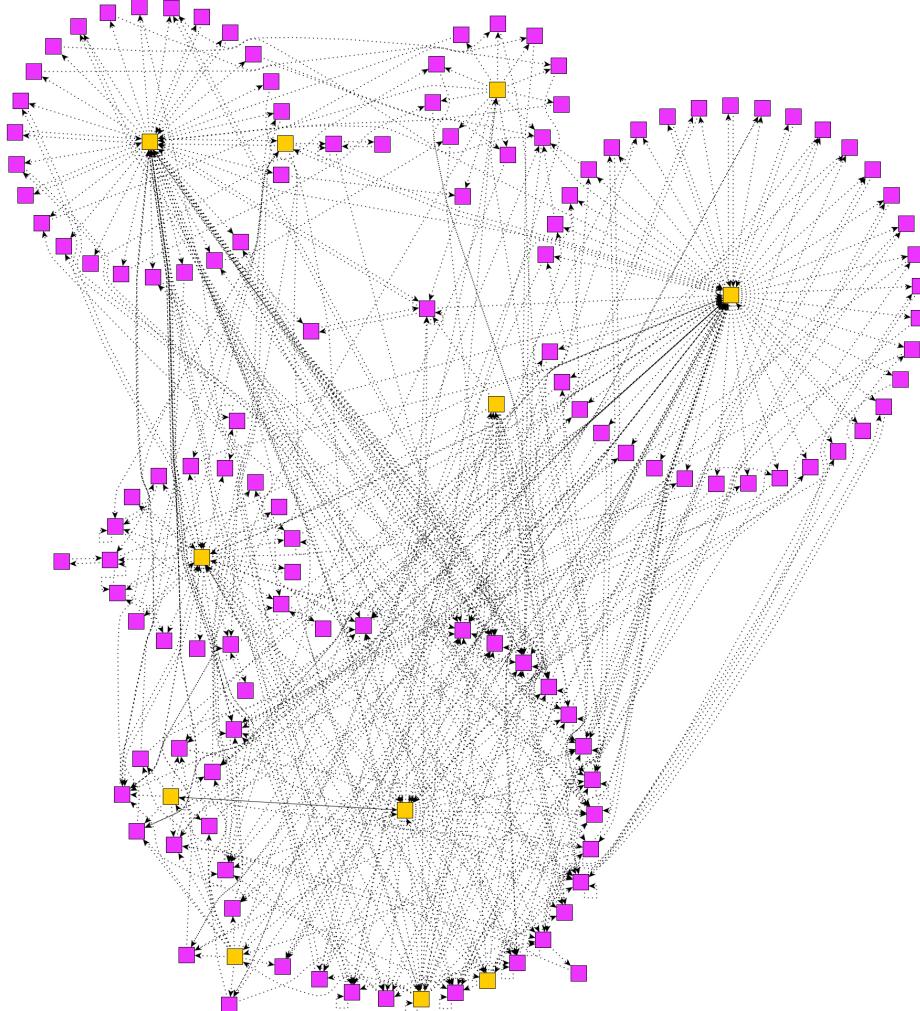


Figure 3.14: AHB Complete Automaton Clusterized

Table 3.1: IMC Coverage Figures (%)

	I <sub>2</sub> C			AHB		
	cov.	tot.	overall	cov.	tot.	overall
Overall	352	410	93.8%	333	1057	61.3%
block	164	180	95.4%	51	68	85.55%
Expression	44	44	100%	7	17	41.18%
Toggle	112	148	75.68%	266	963	30.17%
FSM	32	38	83.36%	9	9	100%

- Nominal 2 - Subgraph made of legal states only, part of the nominal execution.
- Faulty 1 - Illegal state Subgraph, leading to a propagation of the fault to the output.
- Faulty 2 - Illegal state Subgraph, leading to a simulation timeout.

The resulting model is represented on Fig. 3.15.

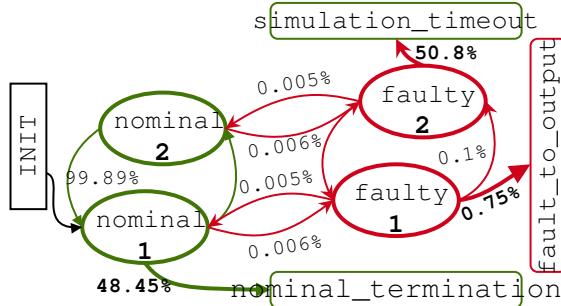


Figure 3.15: Extracted FSM for the I<sub>2</sub>C Block

### 3.6.2 AHB Block Modeling

The AHB bus interface is taken from the GRLIB [30] library with added custom logic to connect it to the master parallel interface of the I<sub>2</sub>C. The added logic comprise an interpreter for the command received by the I<sub>2</sub>C and the glue logic interface to the AHB master side. A verification IP is connected to the AHB slave interface side to respond to transactions and check protocol. Fig. 3.10 represents the translation of I<sub>2</sub>C signals into an AHB transaction by the system. Coverage for AHB block is low and can be explained as only a limited use of the AHB protocol is made:

1. only byte accesses are performed.
2. only single (SINGLE) non-sequential (NONSEQ) transfers are performed.
3. the VIP has not been programmed to insert HREADY wait states in the transaction.
4. the VIP has not be programmed to generate HRESP transaction response error.

The low coverage obtained here doesn't restrict the generality of the methodology but may prevents some failure modes to be identified in this specific case.

### 3.6.3 Complete System Test Case

The complete system is composed of both the I<sub>2</sub>C slave and AHB master along with VIPs at both ends. As previously mentioned, probes are placed on all outputs of the complete system, leaving this time, faults freely propagating internally between the I<sub>2</sub>C and the AHB without being reported by FSV nor the simulation to be stopped. The main difference of this testbench regarding the two standalone previous ones is that faults injected in one block will be able to propagate to the other one (I<sub>2</sub>C → AHB, for example) and back-propagate to the first block (AHB → I<sub>2</sub>C) as simulation will not be stopped when the fault will output from the first (i.e. I<sub>2</sub>C), and later second (i.e. AHB), block. Such "fault loop" (I<sub>2</sub>C ⊕ AHB or AHB ⊕ I<sub>2</sub>C) are expected to be the main possible source of faulty states differences between the standalone and full system faulty states extraction. However, as faults are injected on the inputs in both approach (standalone and full system), we expect to capture, at least a part of thesees "fault loops" induced faulty states in the standalone extractions, if such case exist.

The AltaRica structural model architecture follows the natural hierarchy of the system. As shown on Fig. 3.16, the AltaRica models includes the exact same blocks with the same interconnections between blocks as the functional model. The main I<sub>2</sub>C and AHB modules are composed of two sub-elements, shown respectively in Fig. 3.17 and Fig. 3.18.

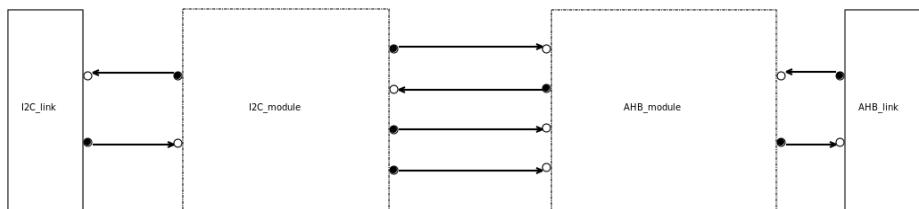


Figure 3.16: I<sub>2</sub>C to AHB System Model

The first element is the *functional* state machine of the module. In case of internal or external fault, this state machine will dispatch the fault to the impacted outputs. This state machine only model the internal faults propagation and do not generate any random failure on its own. The second element is the *internal failure* state machine. This state machine generate internal random failures and provide to the functional state machine the outputs impacted by it. In addition, the AHB module include an additional glue-logic block that converts I<sub>2</sub>C output signals to AHB bridge input signals. No internal failure are generated by this element. At both end of the I<sub>2</sub>C and AHB blocks, links module have been added to model the faults coming from outside of the system.

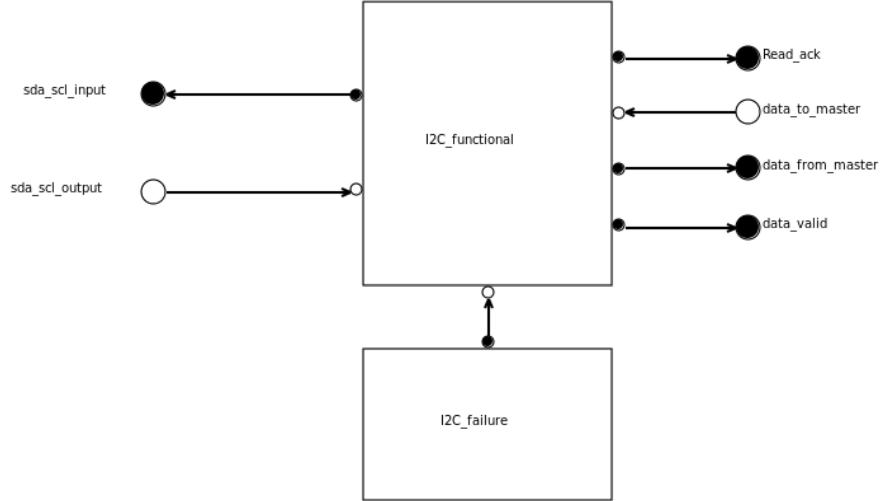


Figure 3.17: I<sub>2</sub>C Block Model

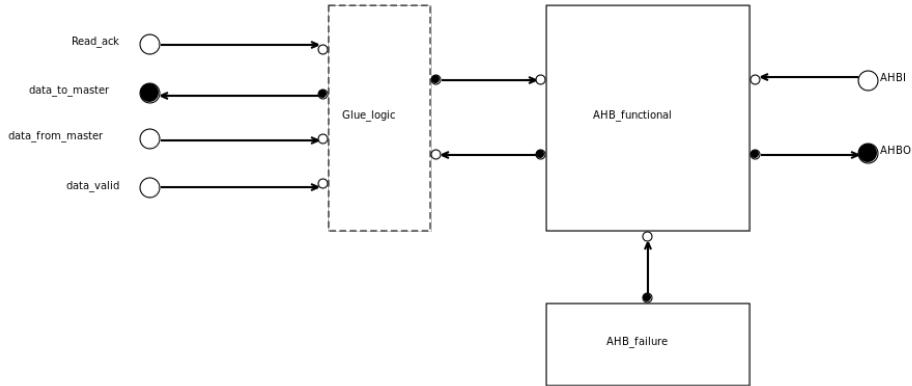


Figure 3.18: AHB Block Model

To model the behavior of the I<sub>2</sub>C and AHB system, only standalone block fault injection test results have been used.

Depending on the methodology, two types of metrics can be extracted. The first one is the probability to propagate internal failure to one or more outputs of the system. From the test results, this probability has been extracted by considering all faults injected in the studied system. The probability to propagate internal faults to an output of the system is then equal to the ratio between the faults detected by the output probe and the total number of faults injected.

The second metrics is the probability to propagate a failure from an input of the system to one or more output of the system. For this metric, only fault

injected on inputs have been taken into account. This probability is the ratio of the input faults leading to an erroneous output over the total number of input faults injected.

SimfiaNeo allows to perform Monte-Carlo simulation. In this type of simulation, a large number of failure scenarios are generated to assess the mean behavior of the system under random failure scenarios. The first possible assessment randomly injected one failure by failure scenario inside the system while the outputs are monitored. If at least one output triggers a faulty state, the error is accounted to have been propagated outside of the system. With this methodology, it's possible to estimate the probability to have a failure propagation from the I<sub>2</sub>C+AHB system to the I<sub>2</sub>C or AHB external signals. The second possible assessment randomly injected one failure by failure scenario in a link module and monitored the other link module. If the opposite link module triggers a faulty state, the error is accounted to have been propagated from one end to another. With this methodology, it is possible to estimate the probability to have a failure propagation from AHB or I<sub>2</sub>C back to the other link.

### 3.7 Results on the *I<sub>2</sub>C to AHB System*

Result of composition obtained by SimfiaNeo are compared to fault injection performed on the full system with probes set only on external outputs of the system on table 3.2 for the I<sub>2</sub>C and AHB side signal. Because the system is simple and faults propagate only forward, it came to simple probability multiplication explaining exact matching of model and system fault injection. No back-propagating faults were observed.

Table 3.2: RTL Fault Injection Vs Altarica Model Failures

AHB Failure	I <sub>2</sub> C+AHB RTL	Altarica Model	I <sub>2</sub> C Failure	I <sub>2</sub> C+AHB RTL	Altarica Model
haddr	0.00551	0.00551	SDA	0.00338	0.00339
hwdata	0.00382	0.00382	Read req	0.00254	0.00255
hsize	0.00068	0.00068	Data	0.00580	0.00581
hbusreq	0.00026	0.00026	Data valid	0.00322	0.00323
hwrite	0.00022	0.00022			

### **3.8 Second Proof of Concept 4BlocksSystem**

### **3.9 Discussion and Future Work**

In this work, we have proposed and experimented the use of Model-Based Safety Assessment on digital system for safety analysis. We have addressed the construction of dysfunctional model for digital system using simulation and have been able to build a simple, but functional dysfunctional model in Altarica. Ongoing work include automatic dysfunctional models reduction to more than one state and the application to a small RISCV SoC and software reliability[28].

# CHAPTER 4

## FOOTNOTES AND SIDENOTES

### 4.1 Introduction

In the last years the density of integration in VLSI systems and microprocessors performances have continuously increased, thanks to the relentless technology scaling. Even though this trend can only continue on its path, several constraints may obstruct the way (power, energy, performance), in particular *reliability* (or cross-layer resilience) can become the more relevant. Hardware redundancy can be used to manage errors at the hardware architecture layer, and eventually even software implemented error detection and correction mechanisms can manage those errors that escalated from the lower layers of the stack [19] [33]. Overall, the goal is to determine the resilience of a particular system in determined conditions, meeting the requirements considering its sensitivity to hardware faults.

It is also true that software failures are not only caused by software implemented faults, as it has been shown [52] the propagation of hardware faults plays a central role, eventually catastrophic. Base on what literature reports on hardware faults evaluation reports [23] [24] it is possible to observe that the percentage of software failure that are caused by pure hardware faults average around 10% [39]. The most famous example is surely the crash of the Mars Polar Lander [7], which cause was established to be dependant on hardware faults resulting in software failure. In that case the lander was not able to settle the legs into their deployed position, which is an hardware fault, and the software gave a wrong order to turn off the engines in the air of Mars, which is a software fault. The system crashed and the entire mission failed.

This paper not only wants to furthermore analyse the behaviour of software failure due to hardware propagated fault but parallelly to the main research [27] path that applies these new methodology to Hardware design in order to

simplify the reliability assessment, the idea of applying the same method in the scope of the assessment of the reliability of software has never been tested. In order to do so, there is the need to specify the main characteristic that Software Products have, fundamental to lay the basis for the described work. Every software can be divided into basic block, atomic chunks of software having the following proprieties:

- One entry point, meaning no code within it is the destination of a jump instruction anywhere in the program.
- One exit point, meaning only the last instruction can cause the program to begin executing code in a different basic block.

Under these circumstances, whenever the first instruction in a basic block is executed, the rest of the instructions are necessarily executed exactly once, in order. The code may be source code, assembly code, or some other sequence of instructions. More formally, a sequence of instructions forms a basic block if:

- The instruction in each position dominates, or always executes before, all those in later positions.
- No other instruction executes between two instructions in the sequence. This definition is more general than the intuitive one in some ways. For example, it allows unconditional jumps to labels not targeted by other jumps. This definition embodies the properties that make basic blocks easy to work with when constructing an algorithm.

The blocks to which control may transfer after reaching the end of a block are called that block's successors, while the blocks from which control may have come when entering a block are called that block's predecessors. The start of a basic block may be jumped to from more than one location. Laid these basis, if, as we'll show in this paper, the reliability metrics extracted for each basic block can be recomposed just knowing the sequence of block required to execute a precise operation, the need for a fault injection campaign on the entire software product doesn't stand anymore.

This paper is organised as follows: the current state of the art is summarised in section II; section III describes the proposed methodology, including its setup, the fault injection procedure and the re-composition of the results from each basic block; a test case is provided in Section IV, while Section V presents the obtained results and sketches some perspectives.

## 4.2 State of the Art

The rush to develop a methodology to assess the reliability and availability of electronic systems has speed up together with the increasing complexity of the microelectronic systems and the miniaturization of such devices. In particular an eye has been kept onto the propagation of faults throughout the entire stack of layers that compose the system as whole, starting from the technological layer all the way up to the software/application layer passing through hardware. In particular the extraction of reliability metrics for software has been the focus of a consistent thread of research [39][38][63] that aimed to verify:

1. whether the software respects the specification requirements,
2. the improvement of the software quality and,
3. *the reliability of the software*

Tools to verify the reliability of software, defined as the probability of the correct software performances for specific period of time on specific environments, have been already developed. In particular the SyRA [64] Cross-Layer Soft Error Resilience evaluation framework proposes a solid method to move from the industrial level Cross-Layer evaluation techniques that are still mainly guided by the sole experience of the designers [19]. These methods are all based on the use of fault injection tools, and they all produce satisfying results in their fields. Nevertheless they have limitation, the description of the Software Fault Models have always been based on the simulation of propagation from the hardware architecture up to software routines, assessing their impact in the correctness of the computation as in [50][66][46]. Moreover no attention has been given to the enormous effort that this type of campaign require, in terms of time, licences for tools and computational power, for an assessment that is limited to the hardware the application is running on and most importantly on the inputs the software receives to perform its calculation. This makes the assessment completely not re-usable in the future requiring a completely new set of campaigns.

Here the focus will be, instead, put on how the software computation reacts to the vulnerable hardware underneath and most importantly to the development of a methodology like there are no other example in the related research, the possibility of decomposing the software products to abstract the single basic blocks and perform a reliability assessment on the single, apparently meaningless blocks to then recompose them obtaining the reliability assessment with a huge time and computational power advantage with respect to the existing methods.

## 4.3 Methodology

The Classical reliability assessment of Hardware as well as Software is Fault Injection driven. The extensive usage of commercial fault injection tools like the ones provided by **Cadence** [15] or **Synopsys** [8] guarantees the proper exploration of the behaviour of the DUT when subject to SEU or other types of faults. This allows the Verification Engineers to have an idea of the behavior of their design without the need to move onto practical testing in radiation environments, which require a dedicated setup [51] and an expensive and not widely available infrastructure.

These advantages come at two main costs, *time* and *Computational Power*, which are consumed in great quantities by the above mentioned simulators. Attempts of Optimization and Parallelization have been put in practice before, but they are not tackling the bigger overhead that we need to take care of every time we simulate a design. Let us assume that, as shown in Fig:4.3 there is the need to test an entire Software Product composed of  $n$  basic blocks, this simulation will last as long as the time to initialize  $T_{init}$  plus the time of the checker/footer to be executed  $T_{foot}$  plus the sum of the duration of all basic blocks multiplied by their multiplicity through the program  $m_n \cdot T_{bb_n}$ . All multiplied by the number of runs that the simulator has to perform to achieve the desired number of injections  $I$ , resulting in:

$$T_{campaign} = I \cdot \left[ T_{init} + T_{foot} + \sum_0^N m_n \cdot T_{bb_n} \right] \quad (4.1)$$

In which the entire program is executed every time entirely, the method proposed by this paper consists in a fragmented study of the basic blocks composing the software, extracting the same metrics that would be extracted by the same fault injection campaign on the whole Software. In this case, in the same way we did before, it is possible to calculate the time needed to carry out the fault injection campaign as we have defined it now, on separate basic blocks, each of them having their random initialization and checker to ensure functionality.

$$I \cdot \left[ T_{init} + T_{foot} + \sum_0^N T_{bb_n} \right] \quad (4.2)$$

In this way we have drastically reduced the amount of time needed to perform the same amount of fault injections, just focusing on the single blocks. Moreover, the difference between the two previously calculated timings, will give us the benefit of studying the blocks singularly, as follows:

$$\begin{aligned}
I \cdot \left[ \sum_0^N m_n \cdot T_{bb_n} \right] - I \cdot \sum_0^N T_{bb_n} &= \\
= I \cdot \left[ \sum_0^N m_n T_{bb_n} - \sum_0^N T_{bb_n} \right] &= \tag{4.3} \\
= I \cdot \sum_0^N T_{bb_n} \cdot (m_n - 1)
\end{aligned}$$

which means that we save the time needed for the execution of each basic block multiplied by its multiplicity, minus one that we still have to execute. Clearly this saved time increases with the length of the Software and therefore the multiplicity of the blocks. In particular, the length of the Fault injection campaign on the entire software is linear with respect to the increasing of multiplicity of the basic blocks, for example due to a larger data input, whereas the solution proposed in this paper is linear with respect to the overall number of unique basic blocks, which remain the same regardless of the data.

### 4.3.1 Setup

The first step towards the application of the method described in the previous section, is the identification and of the different basic block that compose the Software Product under analysis. This can be easily carried out automatically by a simple parser. Basic Block at Assembly level are easy to identify and parse thanks to their intrinsic definition of linear chunks of code. It is therefore trivial to identify in the code all those instructions that modify the flow of the program, tearing down the hypothesis of linearity that defines the blocks themselves. For instance, all the jumping and branching point define the end of a block, as well as the beginning of the following one. Labels in the code also identify starting point of basic block, as they are frequently arrival points for the above mentioned jump and branch operations.

Although having the set of basic blocks divided in single file may seem sufficient, there still the need to initialize all the resources that both the processor and the basic block itself need to run properly, as well as a control logic to ensure that the functionalities of the basic block are preserved (or not) throughout the course of the fault injection campaign. As Shown in fig:2 a **random** initialization is included in the header for the basic block, ensuring the non dependability of the reliability metrics extracted on the input data, together with a footer that checks the functionalities of the block itself. Notice that in this case, contrary

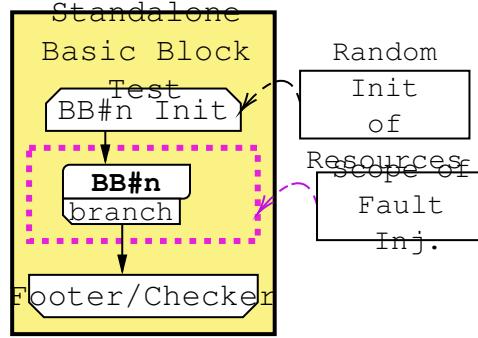


Figure 4.1: Block Diagram of the Entire SW Product

of what is done in the Hardware methodology, there is no physical probing of the circuit on which the program or the testbench is running. In this study only the functional aspect of the Software Product under test is observed.

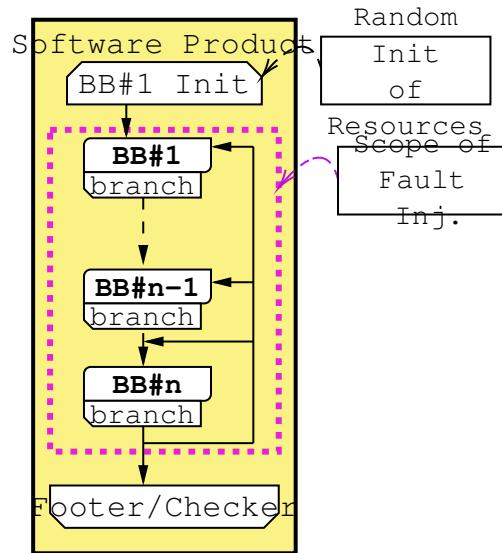


Figure 4.2: Block Diagram of the Entire SW Product

### 4.3.2 Fault Injection on Randomly Initialized Resources

Fault injection is the mean by which the misbehavior and faulty execution is provoked on purpose on digital systems. In the past, especially on hardware, fault injection was aimed to functionally verify the designs under test. Those DUT were analysed, their functions (data dependent) extracted and inputs

were selected in order to exercise those functions. Later on the fault injection had the role of determining whether those functions were preserved in cases of fault or how eventually they were modified. Today this is still the state of the art for software verification.

With time a second approach on hardware was presented, testing moved from functional to structural, where the integrity of the device is evaluated, regardless of the function (and therefore of data), verifying solely the implemented boolean function.

The methodology introduced in this paper presents the novelty of applying this structural approach to software. To abstract the basic block as much as possible from its link to data, **every resource utilized has been randomized** before each fault injection. The probability of failure and propagation probabilities are therefore extracted **independently** of their data input.

Probes (i.e., observation points) are defined during the setup of the fault injection campaign. It is the role of the Footer/Checker (out of the scope of the fault injection) to redirect the output of the block function into a reserved portion of the memory to be probed. Probes are set on those reserved memory location on all blocks, not probing the correctness of the data with respect to the golden run, but solely if the basic function included in that portion of code has been affected by the fault injection.

### 4.3.3 Re-Composition of the basic blocks

Once the fault injection campaigns are over, it is time to re-compose the information that have been extracted on the single blocks into a complete description of the original Software Product. To perform the re-composition there is the need run the software once and record the trace, this will allow us to know exactly the sequence in which the basic blocks have been executed during the nominal run.

We distinguish two main branches of the re-composition, the ones containing fault that *do not modify the program flow* and those that lead to a *modified program flow*

#### Not modified Program Flow

First we nee to define the probability of being executing a precise basic block in time during the execution of the program. Assuming a deterministic duration per executed instruction, without nested or hidden operation, we can define

the probability of executing  $BB_n$  as

$$P_{in-bb_n} = \frac{instructions - in - BB_n}{total - instruction - in - exe} \quad (4.4)$$

Next step is to define the probability of a fault happening in  $BB_n$  being able to become an error in the same block. This has been deduced from fault injection and must be differentiated per every register in which we inject faults and it represented as:

$$P_{gBB_n}^{A_m} \quad (4.5)$$

Last probability to define is the probability of a block to receive a wrong input and propagate it to its output. Defined as:

$$P_{pBB_n}^{A_m} \quad (4.6)$$

which is related to the "time of life" of the variables, defined as the number of basic block between the last time a variable has been read and the first time it gets overwritten.

Once these probabilities ha been defined we can describe the worst possible case, in which a fault is injected in  $BB_n$  and gets propagated throughout the whole program.

$$\begin{aligned} P_{in-bb_n} * P_{gBB_n}^{A_m} * & \left[ P_{pBB_{n+1}}^{A_m} \dots P_{pBB_f}^{A_m} \right] + \\ & + P_{in-bb_{n+1}} * P_{gBB_{n+1}}^{A_m} * \left[ P_{pBB_{n+2}}^{A_m} \dots P_{pBB_f}^{A_m} \right] \dots \end{aligned} \quad (4.7)$$

which summarizes, per every register  $A_m$  as:

$$\sum_{n=0}^N P_{in-bb_n} * P_{gBB_n}^{A_m} * \left[ \prod_{i=n}^N \left[ P_{pBB_{i+1}}^{A_m} \right] \right] \quad (4.8)$$

$$P_{tot} = [P_{in-bb-x} * P_{p-bbx}] + [P_{p-bbx+1} * P_{p-bbx+1}] \dots \quad (4.9)$$

## Modified Program Flow

Regarding the possibility of having a fault injected on a register while the program is executing a precise Basic Block that requires a branching operation at the end, we cannot consider them while recomposing the metrics as in the previous subsection.

These blocks contribute instead to the composition of a particular subset of runs (diverse behaviour of the program) which include all those runs in which the program simulation has reached the end in a time that differs from the nom-

inal one. In particular, it can be shortened due to a premature jump to the conclusive part of the program as well as delayed due to an incorrect loop that sends the machine into a non-necessary series of states from which it will eventually recover. In the case in which the machine would not be able to recover, we categorize those runs as Timeouts (when longer than 150% of nominal time). *It worth to point out that, due to the nature of the injections, which focus on the Register file, with one SEU per run, these cases are reduced to the minimum, if not nonexistent.* Give these assumptions, taking into account this second section of Basic Blocks, it is possible to assume that most, if not all of these runs will generate a failure in the functionalities of the program itself. therefore the recomposition, that was missing a good half of what was neded, now finds the missing cases in all those blocks that led to a modification in the flow.

In particular, considering the possibility that this blocks have not to propagate (to mask) a fault occurring in the course of their routine, the event of flow corruption has probability  $1 - P_{masking}$ , then the probability of these fault becoming a funcional error is 100% and it does not propagate. In this case the recomposition technique is slightly different than the previous section, as the case of a missed branch or jump leads directly to an error. So defined the multiplicity of the same critical block in the nominal sequence  $m$ , the probability of having a functinoal failure is described by:

$$P_{err} + P_{msk} * P_{err} + (P_{msk})^2 * P_{err} \dots + (P_{msk})^m * P_{err} \quad (4.10)$$

Taking into accoun the approximation due to the algorithm intrinsic ability to recover from a flow error.

## 4.4 Test Case and Application

### 4.4.1 The Software

The Software of choice for the Proof of Concept of this methodology is the Bubble Sort Algorithm, in its Assembler for RISC-V Version. Bubble sort is an  $O(n^2)$  sorting algorithm. A simple sorting algorithm that performs a one-way comparison of two adjacent records from the head to tail of the disordered part in each sort trip. Of course, the direction can also be the contrary, one-way comparison from the tail to head of the disordered part. This will form gradually an ordered table at the head of the disordered table, and the basic idea of the algorithm has no difference with the foregoing [45].

Table 4.1: Result of Fault Injection on basic blocks

	x0	x1	x2	x3	x4	x5	x6	x7	x8	x9	x10	x11	x12	x13	x14	x15	x16		x31
bb_0	o	o	139	o	o	o	o	o	379	o	o	o	o	o	o	162	o	...	o
bb_1	o	o	o	o	o	o	o	o	500	o	o	o	o	o	o	o	o	...	o
bb_2	o	o	o	o	o	o	o	o	269	o	o	o	o	28	14	38	o	...	o
bb_3	o	o	o	o	o	o	o	o	500	o	o	o	o	16	131	244	o	...	o
bb_4	o	o	o	o	o	o	o	o	495	o	o	o	o	o	o	62	o	...	o
bb_5	o	o	o	o	o	o	o	o	380	o	o	o	o	o	o	o	o	...	o
bb_6	o	o	o	o	o	o	o	o	494	o	o	o	o	o	o	41	o	...	o
bb_7	o	o	o	o	o	o	o	o	466	o	o	o	o	o	o	o	o	...	o

Table 4.2: Comparison of Fault Injection data vs Recomposed data on Entire Software

	x0	x1	x2	x3	x4	x5	x6	x7	x8	x9	x10	x11	x12	x13	x14	x15	x16		x31
FI	o	o	4	o	o	o	o	o	221	o	o	o	o	15	41	90	o	...	o
Reco	o	o	5	o	o	o	o	o	228	o	o	o	o	13	44	98	o	...	o

Table 4.3: Control Flow Driven Errors

	x0	x1	x2	x3	x4	x5	x6	x7	x8	x9	x10	x11	x12	x13	x14	x15	x16		x31
FI	o	o	o	o	o	o	o	o	198	o	o	o	o	4	77	90	o	...	o

#### 4.4.2 The Division in Basic Block

The processing of dividing the Software under test into basic block has been carried out automatically and returned 8 different blocks, together with the list of resources that each and every basic blocks utilizes during its own functions. After a Nominal run without faults of the entire software, it was possible to trace the transition between the different basic blocks throughout the whole execution. These information, summarized in the scheme below, will be the key to predict the behaviour of the program starting from the behaviour of the basic blocks themselves.

#### 4.4.3 The Platform

The choice of the platform on which the program has been run and tested fell on the SCRiSOC. SCRi is an open-source and free to use RISC-V compatible MCU-class core, designed and maintained by Syntacore. It is industry-grade and silicon-proven (including full-wafer production), works out of the box in

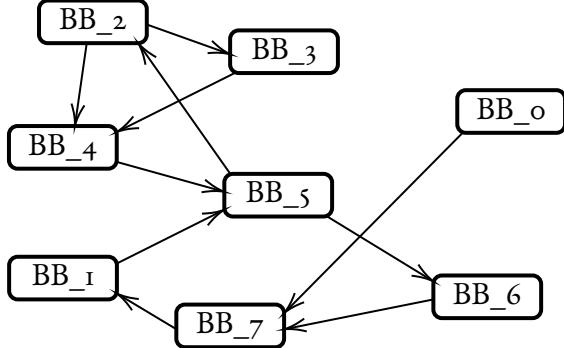


Figure 4.3: Block Diagram of the Entire SW Product

all major EDA flows and Verilator, and comes with extensive collateral and documentation [1]. This choice had mostly been driven by the larger and larger usage of these kind of RISC-V based cores in the academic community. Any test based on these platforms is an added value to their development.

#### 4.4.4 The Fault Injection Campaign

The next step in the methodology is fault injection. It is performed using Cadence fault injection tool *FSV* [15]. Once fault injection sites are automatically identified from the RTL description, fault injection is performed and 20 faults are injected per identified site using a custom pre-generated fault dictionary, including a random injection time. An in-house tool build on top of the *GSL* [31] has been developed for this purpose. Such number is statistically significant enough [41] without compromising fault injection campaign running time. Faults are injected on the integrity of the register file, mimic as well the possibility of faults propagated to the memory and back. Also, fault probes are set on non exercised memory location to record which injected faults will cause a functional failure of the basic block. For each fault injection run, a log-file is generated which reports the outcome of the run, later a custom made parsing tool will recollect the data from this logfile and present the results to the re-composition tool.

### 4.5 Results and Future Work

The results of the recomposition are based on Table:1, which summarizes the result of the fault injection campaign that has been carried out on the single

basic blocks. Each entry of the table enumerates the number of functional error on caused by each register in the register file, keeping in mind that every bit in the register has been affected by 20 faults randomized in time, for a total of 640 faults per register. Once these table has been given to the recomposition tool, Table:2 is returned, including the benchmark fault injection campaign on the entire Software Product for validation of results together with the expected number of faults, calculated following the methodology described. Last, Table:3 Reports the number of Errors that have been caused by an error in the flow of the program, which can be extracted by an equivalent of table number 1 for flow errors caused by each register failing in each basic block and recomposed as in its dedicated section.

The last part of the methodology will be the focus of the work to come, as includes the implicit ability of the different algorithms to recover from flow errors, which understanding can lead to much more refined results.

# CHAPTER 5

## TECHNICAL ANALYSIS

In this chapter, a description of one of the LHC detectors at CERN is given, as to give a better understanding of the environment and setting in which the SoC is planned to function. Following this, a theoretical outline is given for radiation and its effects on CMOS electronics and how this can be mitigated by design choice. Following this, a method for extensive verification of individual block and system verification is presented. After this a description of the steps of physical implementation is given and finally the choice of CPU for the SoC is made, which is accompanied by a description of the state of the system at the beginning of the author's internship and the future plans for it.

### 5.1 LHC Detectors at CERN

There is a vast complex network of different accelerators and different detectors at CERN. The highest energy endpoint in this network is the LHC. The LHC is a 27 km counter-rotating accelerator. Using superconducting magnets, it is capable of accelerating protons up to a peak energy level of 7 TeV, which results in a peak collision energy of 14 TeV. To reach these energy levels, a network of several accelerators is used to initially accelerate the protons to 450 GeV before they are injected into the LHC. In the LHC the beams collide with a bunch spacing of 25 ns corresponding to a frequency of 40 MHz and is called the bunch-crossing (BX) rate. The entire network can be seen in figure 5.1. More details can be read in the article: LHC Machine [26].

## CERN's Accelerator Complex

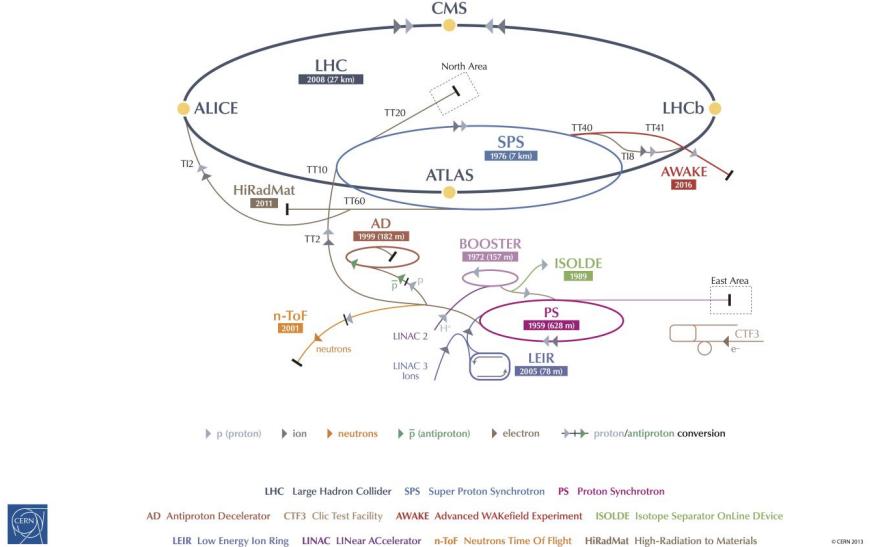


Figure 5.1: Shows the complete network of used accelerators at CERN (Courtesy of [32]).

There are four detectors placed on the LHC ring. These are A Large Ion Collider Experiment (ALICE), Compact Muon Solenoid (CMS), Large Hadron Collider Beauty (LCHb) and A Toroidal LHC ApparatusS (ATLAS). These 4 experiments can be split into 3 categories. CMS and ATLAS are general-purpose onion detectors, which means they try to detect all particles created by the collision. These two are built in different ways by different independent teams such that they can be used to verify the results of each other. ALICE is an experiment focused on the collision of heavier ions, e.g. lead ions. LCHb is a detector focused on only detecting the particles created by collision, which moves in the beam direction. This makes it possible for them to make a specialized detector better suited for detecting the particles in that specific direction. A majority of the particles created in this experiment are related to the beauty quark, which gives reason to its name. A detailed description of CMS, one of the general-purpose experiments, will now be given. This will lay the foundation for understanding the environment in which the electronics are expected to survive and give a reasoning to some of the design choices made later.

The CMS sits at one of the four collision points in LHC. Figure 5.2 shows the layout of the CMS detector. The particles generated in the collisions propagate radially, traversing the silicon tracker. The silicon tracker measures the particle trajectory and transverse momentum  $p_T$ . The silicon tracker is com-

posed of an all-silicon pixel and strip tracker [20]. Next are the electromagnetic calorimeter (ECAL) and the hadron calorimeter (HCAL). The calorimeters enable the evaluation of the particle energy. The ECAL uses lead tungstate scintillating crystals for this purpose [20]. Scintillating crystals emit photons when ionizing particles pass through them. The light is then detected by silicon avalanche photodiodes (APD) in the barrel region and vacuum phototriodes (VPT) in the endcap region. The APD makes use of the avalanche effect, where a single charged particle can knock multiple electrons out of their bond and thereby amplifying their electrical signature. The VPTs are single amplification stage photomultipliers. They have a photocathode at ground potential, a single dynode biased at 600 V, and an anode biased at 800 V. VPTs operate by a photon hitting the cathode which releases electrons. The released photoelectrons are accelerated towards the dynode, where each photoelectron releases multiple new photoelectrons. These are then accelerated towards the anode as it is at a higher potential. The anode then produces an amplified current.

After the ECAL, the particles enter a brass/scintillator HCAL [20]. Here the scintillation light is collected by wavelength-shifting (WLS) fibers embedded in the scintillator tiles. The WLS fibers emit multiple low-energy photons for each high-energy photon strike. This light is channeled to photodiodes which amplify the signal. The aforementioned components are encapsulated by a 3.8 T superconducting solenoid. Outside the superconducting solenoid, the iron return yoke with muon chambers is placed. The iron return yoke confines the magnetic field and stops all remaining particles except for muons and neutrinos. The muon system has 3 functions: muon identification, momentum measurement, and triggering. In the barrel, region detection is done using drift tubes while in the end-cap region it is done using cathode strip chambers. Both of these systems are completed by a dedicated trigger system of resistive plate chambers.

In total, the CMS detector has a diameter of 15 m, a length of 28.7 m, and weighs  $14 \cdot 10^6$  kg.

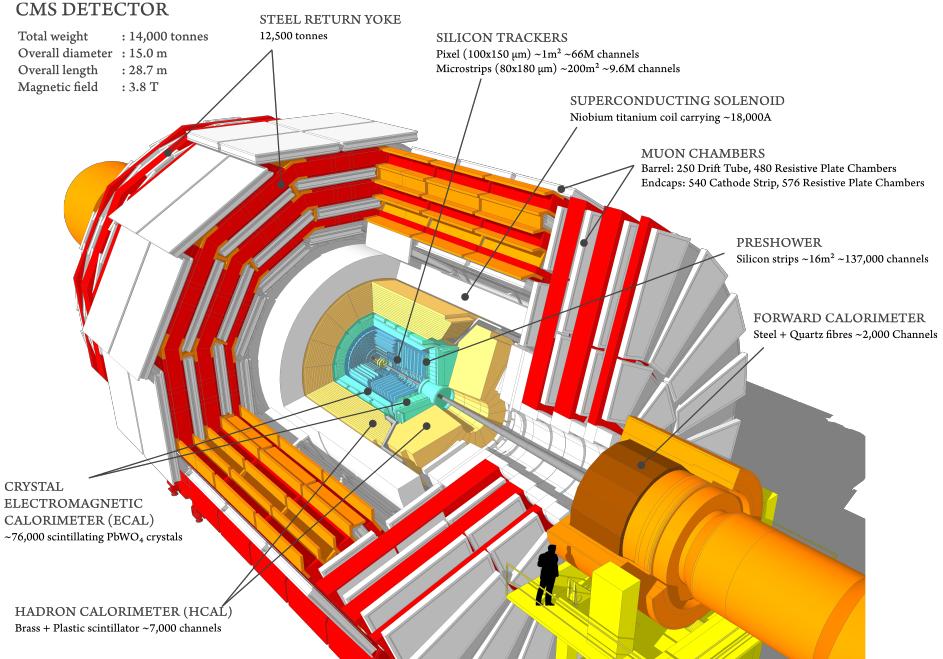


Figure 5.2: Shows the layout of the CMS detector [17].

The CMS is capable of detecting a wide range of particles using the collection of the data from each of its components. Different particles will follow a different path through the detector based on their charge, momentum, and trajectory. In figure 5.3 a path of the common particles can be seen. The superconducting solenoid enables the estimation of the charge and momentum of a particle. The charge can be determined by the bend direction of the trajectory because positively charged particles will bend opposite to negatively charged particles and neutral particles will not bend at all. The momentum can then be estimated by the degree of bending as faster-moving particles will bend less than slow-moving particles.

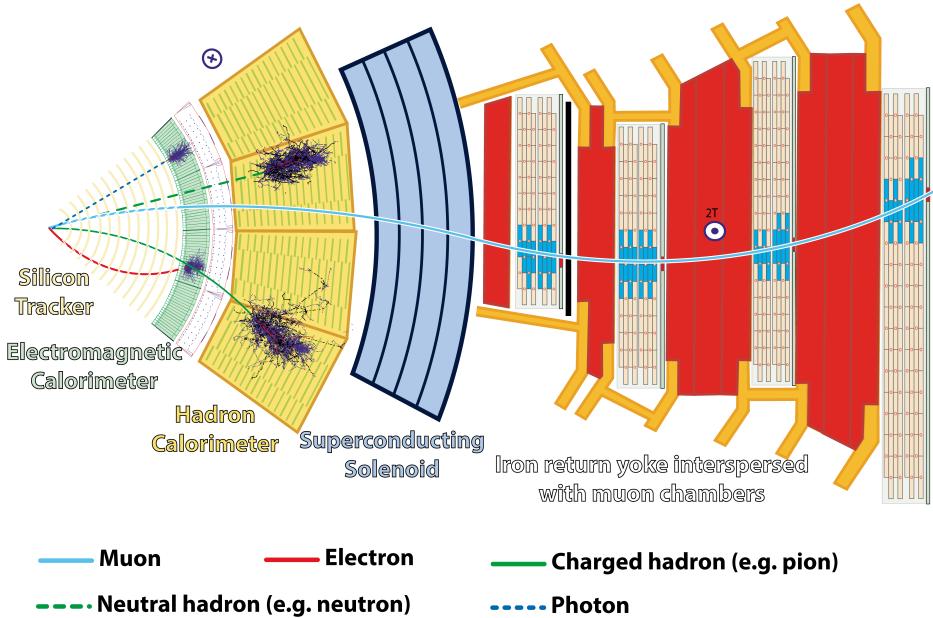


Figure 5.3: Shows the movement of different particles in the CMS detector [5].

The collision of charged particles in the LHC creates ionizing particles which over time will accumulate. The total ionizing dose (TID) expected after 10 years of operation has been simulated using FLUKA, a tool for monte carlo simulation of particle movement and their interactions. The expected dose decreases with distance from the collision point. A complete map of the expected TID for the detector can be seen in figure 5.4.

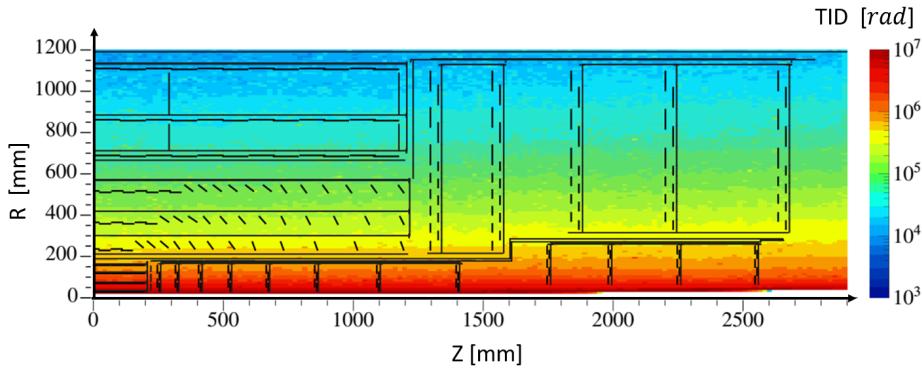


Figure 5.4: Shows expected total ionizing dose in Gy during a 10-year operation period of the CMS. This is simulated using FLUKA. [21]

The ionizing dose and charged particles passing through electronics can alter their behavior and affect the output in unwanted ways. These effects will now be discussed in detail.

## 5.2 Radiation Effects on CMOS Electronics

It is necessary to discuss and understand the effects of radiation on CMOS electronics due to the highly radioactive environment of the CERN accelerators. This understanding will lead to understanding the necessity and the methods for radiation hardening of the electronics. The radiation affects the CMOS in two distinguishable ways. In the form of cumulative damages and single-event effects (SEEs).

### 5.2.1 Cumulative Damages

Cumulative damages can be split into two subcategories. The first is non-ionizing processes, which come in the form of displacement of atoms in the lattice structure of the transistor. These are called displacement damages and are of little concern to CMOS technologies due to the high amount of doping [8]. The second is damages induced by ionizing doses, i.e. TID effects. MOS transistors can accumulate charges in the gate oxide, which creates a voltage difference on the gate and leads to unwanted biasing. This effect is dominant when the gate oxide is thick as it can contain a larger charge compared to the voltage threshold of the gate. Therefore, this effect decreases with smaller technologies as the gate oxide becomes thinner. The smaller technologies are instead dominated by effects like shallow trench isolation (STI) effects, which come in the form of radiation-induced drain-to-source leakage current and radiation-induced narrow channel effects (RINCE) [8]. Radiation-induced drain-to-source leakage current is caused by the accumulation of positive charges in the STI, which opens parasitic channels between the source and drain. This leads to an increase in leakage current. Positive charges are far more likely to be trapped due to electrons moving fast enough to leave the STI, while electron holes do not. However, over time electrons are attracted and enough electrons can be attracted to invert this effect. Therefore, initially, an increase in leakage current is observed, but as TID increases this effect reaches a peak and begins to invert. Since only positive charges are initially trapped, this effect does not increase the leakage current of pMOS. Instead, it repels the holes of the doped silicon increasing its threshold voltage and decreasing current flow. It is clear that increasing the length of the channel, decreases this effect as more charge is to be trapped before a channel can be opened. Therefore this effect is significant in smaller technologies with short gate lengths. However, this effect can be mitigated by using enclosed layout transistors (ELT), where the channel does not face the STI.

The other effect is RINCE, which is also due to the trapped charges in the STI. As positive charges are trapped in the STI, an electric field is created. This electric field leads to a decrease in threshold voltage for nMOS transistors and an increase in threshold voltage for pMOS transistors. However, as the width of the channel decrease, this effect becomes more dominant as the number of trapped charges does not change. This leads to a proportionally larger electric field, which signifies a dependency on channel width for this effect. For nMOS transistors, this effect is limited, as negative charges become trapped at the interface leading to the two canceling out similar to the inversion seen in radiation-induced drain-to-source leakage current at higher TID. However, for pMOS the trapped charges at the interface are also positive, leading to RINCE only increasing in potency with an increase in TID. As this effect is also due to charges trapped in the STI, it can be mitigated by the use of ELT [8]. However, these ELTs do use significantly more area compared to traditional designs.

### 5.2.2 Single-Event Effects

Single-Event effects can be split into two categories, permanent single-event effects, and single-event upsets (SEUs) (and transients (SETs)). A permanent single-event is the possible creation of parasitic transistor structures between two n-wells, i.e. between two transistors. This can potentially shorten VDD and ground, which can permanently damage the device. However, this effect is limited due to highly doped substrates and the use of STI between wells [16].

SEUs and transients are soft errors and not destructive to the die. Instead, they corrupt the information stored in digital logic circuits by flipping bits. SEUs become possible when the collected fraction of the charge liberated by an ionizing particle is larger than the electric charge stored on a sensitive node [16]. This critical charge scales with the gate area of the design. As the gate area decreases, the amount of stored charge representing a logical value of information decreases. In general, SEU sensitivity is increased by the scaling down of technology as node capacitance and the supply voltage are both scaled down as well [16].

A SET is an event, where a static combinatorial circuit is upset by a charged particle, leading to a glitch in the circuit. The time duration of SET is determined by the injected charge and the driving strength of the cell. If the output of this combinatorial circuit is sampled during the transient, a register can enter a metastable state, which can propagate through causing fatal errors [16].

### 5.3 Radiation-Tolerant Design

The goal of radiation-tolerant design is to limit the potential damage caused by radiation effects as described in section 5.2. Some of these effects can only be limited by a careful layout of the die or by the intrinsic properties of the chosen CMOS technology. However, SEUs can be reduced and mitigated by a combination of digital design. The methods for radiation hardening are usually dependent on redundancy either in space or in time.

A common method is the use of triple module redundancy (TMR). In this method, each memory element (i.e. register) and the corresponding combinatorial logic are instantiated three times. The outputs from these three registers are then passed to a voting system, which outputs the majority vote. The voting system itself is also triplicated to minimize the chance of a SET upset. If a single voter is used and is sampled during the SET, an error will occur in the system, thus leading to a single-point of failure. To avoid the build-up of errors in the 3 different paths, the feedback loop of a state machine should be taken from the voted result. An example of a TMR radiation hardening can be seen in figure 5.5.

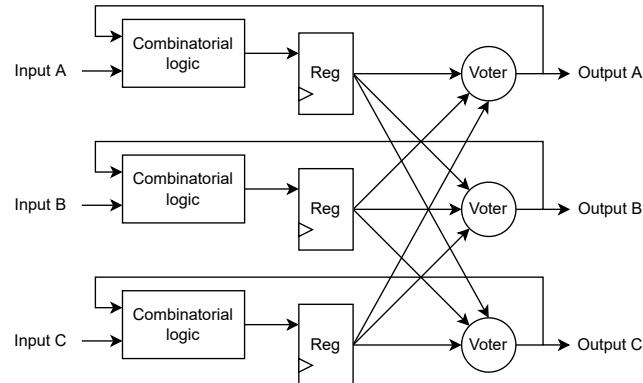


Figure 5.5: Shows a spatial radiation hardening technique using triple module redundancy.

This added redundancy would lose many of its radiation hardness benefits if the triplicated registers are placed close together on the chip die as this would increase the chance of an SEU happening on multiple registers from the same charged particle. Therefore the placement of these registers is restricted in the physical layout such that a minimum distance is enforced. From this, it is clear that TMR increases radiation hardness by using space and power consumption (increased due to increase in hardware) as a trade-off. This increase in power and area is not cheap as the heat generated needs to be transported away from

the detector and the space itself is limited inside the detector. Therefore to limit the disadvantages, the TMR is usually only done to the control path of a state machine. This is done as errors in the data path are limited in time, while an error in the control path can result in complete failure of the chip. At CERN, a tool has been developed for this method of radiation hardening named TMRG.

Another way of radiation hardening is the use of temporal spacing and is done by delaying the clock signal. The registers between combinatorial logic are triplicated, while the combinatorial logic itself, is not. Instead, the clock signal for each of the three registers is delayed, such that the SEU has a high statistical probability of having passed. This results in the SEU only affecting one register. It is then possible to use the same voting system as in TMR to achieve a corrected output. An example of this can be seen in figure 5.5.

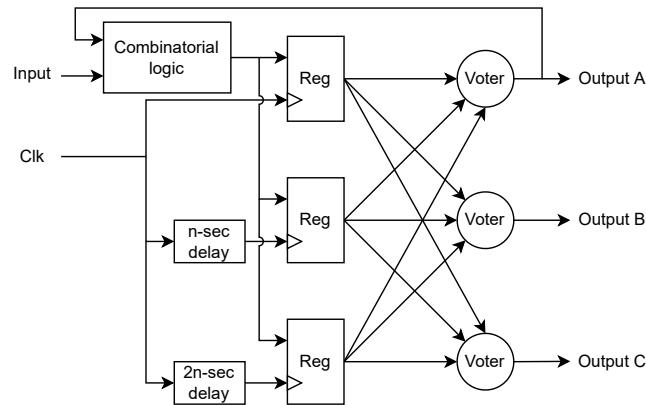


Figure 5.6: Shows a temporal radiation hardening technique where clock signals are delayed.

This method does not have the same minimum distance requirement as the TMR hardening technique as the registers sample at different timestamps. Instead, there is a temporal spacing requirement. This method does not require a triplication of the combinatorial logic and therefore saves on space and power consumption. However, it does make the timing analysis and closure difficult and this only gets more problematic as the frequency increases.

Even though temporal radiation hardening has multiple advantages in the form of space and power consumption, the TMR is chosen due to its simpler implementation. This is due to the problematic nature of timing closure for the temporal radiation hardening, but also due to the existence of an already-developed tool for performing TMR.

## 5.4 Universal Verification Methodology

The universal verification methodology (UVM) is an IEEE industry standard for the verification of design components [2]. It is developed by the Accellera group and its members. The goal is to create a modular, scalable and reusable generic verification environment. For these reasons, this methodology will be used for the verification of the SoC and its components. A short description of UVM will now be given.

UVM is based upon a hierarchy structure laid out by Accellera. This specifies guidelines for the creation of a verification environment and gives the support structure for this development. It does this by supplying a framework, for the designers to build on top of. This framework has the most general and essential features (Reporting, handshake mechanisms etc.), such that they do not need to be redeveloped for each project. This also ensures uniformity in test-bench creation across many different work groups. The framework hierarchy is seen in figure 5.7 as it is laid out by Accellera [1]. Here the UVM agent can be expanded as it contains a sequencer, a driver, and a monitor. The expanded UVM agent can be seen in figure 5.8. Even though this is the recommended structure and should fit the most common use cases, the framework can be customized.

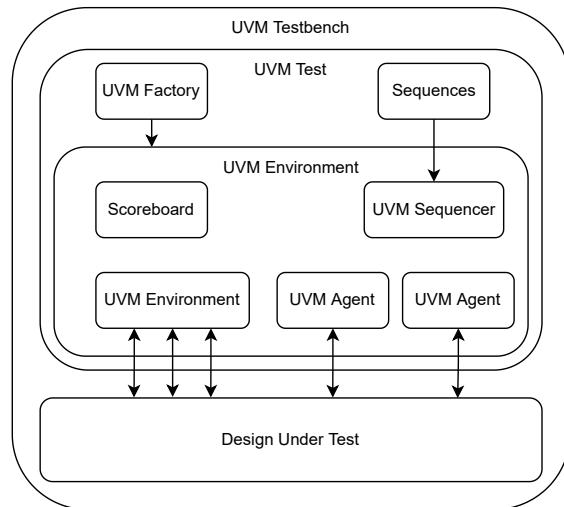


Figure 5.7: Shows the complete hierarchy of the UVM structure laid out by Accellera.

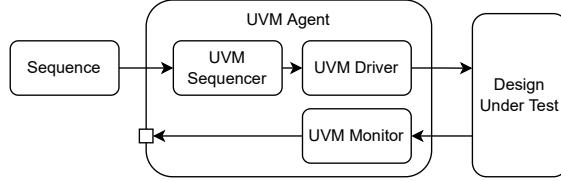


Figure 5.8: Shows the UVM components in a UVM agent and its connection to other components.

The communication between components is based on transaction-level modeling (TLM) and UVM items. The items are designed to fit the specific device under test (DUT). It contains the information necessary to create stimuli to the DUT and updating scoreboard and reference module. The item is sent between components using TLM.

Each of the hierarchy levels has a base class associated with them. It is on top of these base classes that the project-relevant components will be built. Each of these classes and their functionality is described below [1].

- **Testbench:** is the root class and container for all that needs to be simulated and tested. Typically this instantiates the DUT and the connections between the test and the DUT.
- **Test:** is the top-level UVM component. It has 3 main functions. To instantiate the test environment, configure the environment via a configuration database or factory overrides and apply stimulus to the DUT via the UVM sequences. This enables the designer to not have multiple instances of the same environment with different configurations for different test cases. Instead, the environment can be configured from this top-level UVM component to perform those test cases without repeating code.
- **Environment:** is a UVM component that instantiates and contains other reusable verification components such as agents, scoreboards, and other environments. It is also here the different components are connected and configured for default use.
- **Scoreboard:** is the verification component that compares the DUT to an implemented reference module. The scoreboard does this by receiving UVM items from the DUT via the UVM agent using TLM ports. It can then use the reference module as a predictor and compare that to the DUT.

- **Agent:** is a hierarchical component that contains other UVM components. These are typically a sequencer, a driver, and a monitor. The agent can either be active or inactive. An agent is active when it can drive stimuli to the DUT, which requires the use of a driver and sequencer. An inactive agent only contains a monitor.
  - **Sequencer:** controls the flow of sequence items from the multiple sequences to the driver, i.e. queues different sequence items according to a set of given parameters.
  - **Driver:** receives sequence items from the sequencer and converts them from transaction-level stimuli into pin-level stimuli for the DUT. For example, it can take a parallel data packet and transmit it via input pins to the DUT using a specified protocol.
  - **Monitor:** samples the DUT interface to convert data from pin-level stimuli into transaction-level stimuli. This data is then broadcast to the rest of the UVM testbench. The monitor can perform some levels of processing internally. For example, it can receive pin-level stimuli and decipher them according to a chosen protocol. So instead of broadcasting all pin-level activity, it first converts it into a specified UVM item and then broadcasts that using TLM.
- **Sequence:** makes up the core stimuli of the verification plan. A sequence can be made of multiple data items which can be used to create the scenarios for testing the DUT extensively. The items are eventually sent to a sequencer which will then queue it and send it to the driver. Multiple sequences can be connected to the same sequencer. The randomization tools in UVM are commonly used in creating data items such that there is a higher chance of discovering bugs. A sequence is not part of the component hierarchy. A sequence can contain other sequences, called a parent or virtual sequence.

A UVM environment can be developed either in SystemVerilog or SystemC, most commonly SystemVerilog. The SystemC variant is still under development but is in working condition. In this project, it has been chosen to use SystemC.

## 5.5 Physical Implementation of Digital ICs

The PicoRV32 SoC will be implemented in a physical design. Therefore, a description of the steps from a digital design to a physical implementation will

be given. The steps of physical implementation can in general be split into two categories: synthesis and implementation. Doing synthesis a Verilog design is converted from behavioral modeling into a gate-level description using only basic components supplied by a library. In the implementation, the gate-level description is converted into a physical implementation on a die with power distribution, a clock tree, and non-ideal components. Doing most steps, setup and hold conditions are checked for violations using different conditions. These are commonly referred to as corners. Corners are on-chip variations that alter the expected behavior of digital circuits and commonly include process, voltage, and temperature (PVT). The process variations are often summed and split into 3 categories slow (S), typical (T), and fast (F) for nMOS and pMOS transistors. 5 combinations (TT, SS, FF, SF, FS) of these corners can be evaluated to ensure that system works given variations and uncertainties. Setup and hold are tested in these corners, to ensure that the system behaves correctly even given on-chip variations. At CERN, radiation is included as an additional variation parameter. The TID effects are also evaluated in corners, using models developed at CERN for the behavior of chips doing radiation.

### 5.5.1 Synthesis

Synthesis is split into 3 steps: synthesis, mapping, and optimization. Doing synthesis a netlist is created from the supplied behavioral HDL model. For example, the always blocks in Verilog are converted into a basic set of components, e.g. flip-flop, AND-gate, NAND-gate. This step requires a list of constraints supplied by the designer. These constraints can be the clock period, output load capacitance, the maximum transition time of components, and setup/hold uncertainty on the clock. When this step is done, the Verilog code has been converted into a gate-level netlist that implements the same logic.

The next step is mapping. Doing mapping a gate-level netlist is converted from using generic components to using components given by a library called standard cells. This library contains detailed descriptions of common cells in an implementation technology. The technology used in this project is 28 nm. Many libraries exist for the same technology, but with different characteristics, e.g. track number, cell width, and gate length. The number of tracks refers to the height of each cell. The cell width is the number of which each cell width has to be a multiple of. Three libraries will be used as the default for physical implementation. The libraries will be 9 tracks high, a multiple of 140  $\mu\text{m}$  wide, and have a gate length of 35 nm. The three libraries will have different threshold voltages. One will have a standard threshold, one will have an ultra-high threshold voltage and one will have a low-threshold voltage. The threshold

voltage controls the voltage needed for the transistor to switch state. This is controlled by the amount of doping on the source and drain of the transistor. However, lowering the threshold voltage increases the leakage current of the transistor. Different threshold voltages are therefore a trade-off between speed and power consumption. By including multiple libraries, the algorithms are given more options for optimization, such that speed requirements can be met by using low-threshold transistors or power consumption can be reduced by using ultra-high threshold transistors. The libraries contain a description of the physical layout of the cell, a timing model describing the delay from input to output, and noise models. After mapping the gate-level netlist described using generic components has been converted into a gate-level netlist with standard cell components from a specific library.

After mapping, the last step of synthesis is optimization. Doing this step, different optimizations of the gate-level netlist are performed. This can be the removal of grounded circuits, removal of registers containing constants, or re-location of registers to reduce the amount needed. After all the steps have been completed estimates of certain parameters can be given for the digital design. This is things such as worst negative slack, power consumption, number of cells needed, and size of the complete design.

### 5.5.2 Implementation

Implementation is the next step after synthesis. The implementation converts the gate-level netlist into a physical layout on a die. The steps of this are floorplan, place, clock tree synthesis, route, design finishing, and verification.

Doing floorplanning, the die layout is designed. One of these variables is the die size. A rule of thumb is that the estimate at the end of synthesis should be 60% of the final size. However, this does not account for the size of the power rings. These are chosen to be 10  $\mu\text{m}$  wide for both ground and VDD. This results in a spacing of 25  $\mu\text{m}$  of every side from the edge to the area in which standard cells can be placed (Extra space is needed to allow spacing between GND and VDD rings). Power rings are used to ensure a uniform distribution of power to all parts of the chip and to reduce the effect of IR drops. As the height of a standard cell is given by the library, the die is divided into rows with that height. The top and bottom of these rows can then be connected to the power by having wires going across from side to side. This creates a power mesh that supplies all the rows of standard cells with VDD and ground connections. On large designs, these lines can become long. Therefore, to minimize IR drop vertical and horizontal larger stripes are used to connect these tracks at multiple points. In the floorplanning, the height of the die is also given in metal tracks.

For this project 9 metal (M<sub>1</sub>-M<sub>9</sub>) layers will be used. These do not have the same width. Instead, the top layer M<sub>1</sub>, has the highest routing width, M<sub>2</sub> - M<sub>6</sub> has the same width but is lower than M<sub>1</sub>, and M<sub>7</sub>-M<sub>9</sub> all decrease in routing width as the metal layer number goes up. M<sub>1</sub> and M<sub>2</sub> will be used for power distribution as these layers have the largest routing width and therefore the lowest resistance in the wires. M<sub>9</sub> is used for the placement of standard cells and wiring. Usually, it is preferred to use the lower layers for wiring between standard cells as these signals do not draw significant current and the resistance is therefore not of concern. After this I/O ports are placed and if it is a top-level design, pads are also placed.

The next step is placement. Here the standard cells are placed in the rows. This is with the goal of achieving minimal congestion and the best foundation for timing closure. This is typically done by keeping connecting cells close together. The tool also has to do the placement with the minimum distance between triplicated registers.

The next step is clock tree synthesis. In this step, all the relevant components are connected to the clock signal. However, this is not a straightforward process. If the clock was just connected to all components without buffering, the clock edge would arrive at slightly different times depending on the placement of the component. Therefore the clock signal is continuously buffered to increase driving strength on long wires, but also to introduce delay in short wires, so that they arrive at nearly the same time, with the goal of not violating setup and hold time. This creates a tree-like structure. However, there is also a disadvantage to having all flip-flops latch at the same point in time, as this creates a large current spike which might cause IR drops and cause faults. Therefore if there is room in the setup and hold time, it is also an advantage to space out the latching in time, so that the current spike is limited.

After clock tree synthesis, the next step is routing. This is done after clock tree synthesis because swapping these two steps would mean that the clock tree would have suboptimal wiring due to possible high congestion by routing. In the routing step all the connections specified by the gate-level netlist are established. If these wires are long, buffers are inserted to increase driving strength.

The design has now been placed and routed. Power and clock distribution have been established and in theory, the physical implementation process is finished. However, a few steps still remain to verify the design and ensure manufacturing is possible. For example, metal is filled into empty areas of the die. Without this sinkholes would be created and uneven metal layers would be laid on top, this would only get worse as more and more layers are added if nothing is routed or placed in that region. Therefore, metal is placed to create an even

surface on which the metal can be laid. Next, design rule checks (DRC) and layout vs schematic (LVS) are performed. DRCs are manufacturing rules and safety protocols to ensure that the layout can actually be manufactured by the foundry. It can be simple things such as wire proximity checks, no short circuits, etc. However, there are also many complex rules. Things such as the area of a wire on a specific metal layer might be so big that it collects enough charge during manufacturing that it destroys the connected component (Antenna effect). Another example is the concept of electromigration. When current is transported through a wire, it will slowly move material around. This effect increases with the amount of current. If this effect is too large, it means that some parts of the wire might become shallow and thereby increase resistance changing the behavior of a sensitive circuit. It can also short-circuit to a nearby wire if it gets too wide at places. It, therefore, checks if a wire is in danger of changing properties over a given lifetime. LVS checks whether the resulting layout at the end of the physical implementation is equivalent to the digital description from which it was derived.

With the physical implementation complete, we can generate an SDF file. This file contains all the propagation delays, clock arrivals, and such for the gate-level netlist. This can be used in combination with a testbench and the circuit can be verified to ensure that even given non-ideal clock signals and propagation delays, it still behaves as expected. Using this testbench and SDF file, a more accurate estimate of the switching activity of each transistor can be derived. This can be used in a dynamic power analysis to obtain a far more accurate estimation of power consumption. Before this, the tool just assumed that all transistor switches with a 20% probability on each clock edge. This is the final step of physical implementation. All of this can then be collected in a set of files to establish a library for the implemented block. This can now be manufactured or used as a component in a larger design.

## 5.6 The PicoRV32: System on Chip

The goal of this project is to make a simple demonstrator SoC, with a CPU, memory blocks, and standardized interconnect bus with a basic set of peripherals. This demonstrator chip will then be used to test whether it is possible to create a radiation-tolerant SoC with a reasonable amount of area and power usage. Computing power is not of concern as its use case is targeted toward control and monitoring. RISC-V is an open-source instruction set architecture (ISA) that perfectly fits this application. Many different open-source SoC foundations have been considered and 3 stood out as candidates. Those are the Rocket

Chip from UC Berkeley, the PicoSoC, and the Pulpissimo by ETH Zurich. The Rocket chip is a promising chip-building framework in Chisel. The PicoSoC is simple and written in pure Verilog, making it easier to understand. The Pulpissimo is made by ETH which has been collaborating with CERN before and therefore enables more direct communication with the designers. These three have been compared on their power and area usage during post-synthesis. The result of this can be seen in figure 5.9. Based on this comparison, the PicoSoC is chosen for the demonstrator chip.

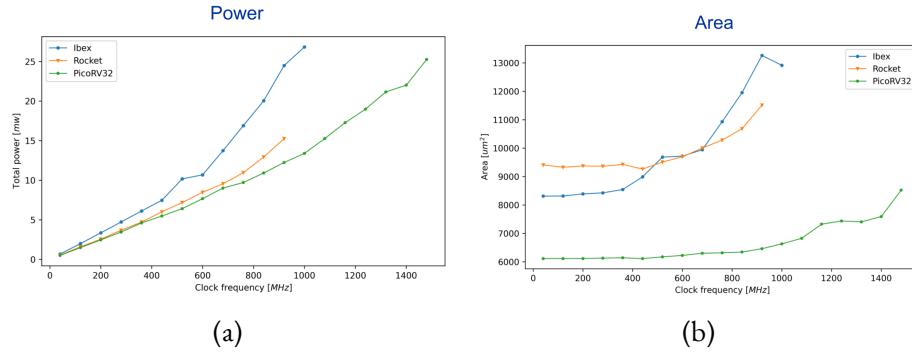


Figure 5.9: Shows the post-synthesis power and area comparison of an Ibex, Rocket, and PicoRV<sub>32</sub> core.

The PicoRV<sub>32</sub> SoC is based upon the RISC-V open-source CPU: PicoRV<sub>32</sub>. This core is meant to be used as a size-optimized auxiliary CPU in an FPGA or ASIC design. It does not have high computational power, but it is small and simple. Due to its simplicity, it is also easier to debug and develop extra features.

RISC-V is an open-source instruction set architecture (ISA). Its architecture is developed on Reduced Instruction Set Computer (RISC) principles. This is in contrast to the Complex Instruction Set Computer (CISC), to which the commonly known family of x86 ISAs belongs. The two design topologies differ because a CISC instruction often executes several lower-level instructions, while a RISC architecture does not. RISC-V employs a base set of the most needed instruction, while several extensions are available to expand the instruction set. The PicoRV<sub>32</sub> is configurable [70]. Its instruction base can be based on either RV<sub>32I</sub> (32-bit, base integer instruction set) or RV<sub>32E</sub> (32-bit, base integer embedded instruction set). The RV<sub>32I</sub> is capable of having two extensions added: multiplication (M) and the compressed instruction set (C). There is also an optional built-in interrupts controller. However, this is not based upon the RISC-V standard and is instead custom-built for the PicoRV<sub>32</sub>. This design choice was made because the RISC-V interrupt handle was extensive

and comprehensive, so a simpler IRQ handling with less hardware overhead is available.

At the start of this project, several things were already implemented: The core, two buses, a bridge connecting the two busses, and a temporary memory block. A Native Memory Interface (NMI) connects the core and memory. The NMI is connected to a bridge that is connected to an Advanced Peripheral Bus (APB) interface. Peripherals will be connected to SoC using the APB interface. The state of the SoC at the beginning of this project is visualized in figure 5.10.

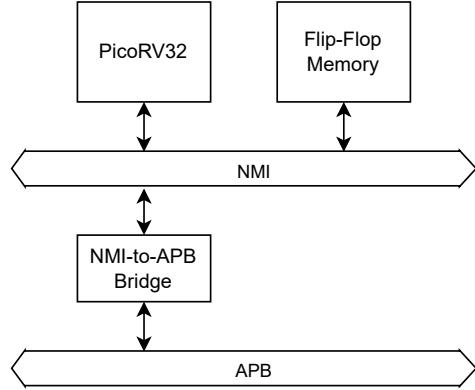


Figure 5.10: Shows the state of the PicoRV32 SoC at the beginning of this project.

A detailed description of the two interfaces used on the buses is now given.

### 5.6.1 Native Memory Interface

The NMI is an interface defined by the PicoRV32 [70]. It is a simple valid-ready interface bus. It requires five outputs from the PicoRV32 core and two outputs from the slave, which is receiving the transfer. These signals and their functions are described in table 5.1. This bus is only used to connect the most essential and critical blocks to the core, e.g., memory and bootloader.

Table 5.1: NMI signal descriptions

Signal	Source	Description
CLK	Clock source	Clock. The transfer is completed on the rising edge.
VALID	PicoRV32 core	Valid. The core uses the valid signal to initiate a transfer. All core outputs are stable while valid is high.
INSTR	PicoRV32 core	Instruction fetch. Used by the core to indicate if the memory transfer is an instruction fetch
READY	Slave interface	Ready. Asserted by the slave when the read data is available and used to acknowledge a write transfer.
ADDR	PicoRV32 core	Address. The core supplies the address which is used by the slave to read or write to the requested cell.
WDATA	PicoRV32 core	Write data. If a write transfer is being performed, the core supplies the data to be written using this bus
WSTRB	PicoRV32 core	Write strobe. If the write strobe is 0, it indicates a read transfer, while it being non-zero indicates a write operation. The write strobe signal is used to write specific bytes of the wdata. It is possible to write 32 bits, the upper 16 bits, the lower 16 bits, or 8 bits.
RDATA	Slave interface	Read data. In the case of read transfer, this is the data read from the specified address. When rdata is available, ready is asserted.

This bus is fully triplicated to achieve radiation hardening. This is decided due to it being critical infrastructure and also the length of this bus being limited since peripheral devices are not connected to this bus.

### 5.6.2 AMBA APB Interface

The APB is designed by ARM and is part of the Advanced Microcontroller Bus Architecture (AMBA) protocol family. This protocol has been chosen for the SoC since it is designed for minimal power consumption and reduced interface complexity [3], which aligns with the goals of the project. The list of signals in the protocol can be seen in figure 5.11.

Table 2-1 APB signal descriptions

Signal	Source	Description
<b>PCLK</b>	Clock source	Clock. The rising edge of PCLK times all transfers on the APB.
<b>PRESETn</b>	System bus equivalent	Reset. The APB reset signal is active LOW. This signal is normally connected directly to the system bus reset signal.
<b>PADDR</b>	APB bridge	Address. This is the APB address bus. It can be up to 32 bits wide and is driven by the peripheral bus bridge unit.
<b>PPROT</b>	APB bridge	Protection type. This signal indicates the normal, privileged, or secure protection level of the transaction and whether the transaction is a data access or an instruction access.
<b>PSELx</b>	APB bridge	Select. The APB bridge unit generates this signal to each peripheral bus slave. It indicates that the slave device is selected and that a data transfer is required. There is a <b>PSELx</b> signal for each slave.
<b>PENABLE</b>	APB bridge	Enable. This signal indicates the second and subsequent cycles of an APB transfer.
<b>PWRITE</b>	APB bridge	Direction. This signal indicates an APB write access when HIGH and an APB read access when LOW.
<b>PWDATA</b>	APB bridge	Write data. This bus is driven by the peripheral bus bridge unit during write cycles when <b>PWRITE</b> is HIGH. This bus can be up to 32 bits wide.
<b>PSTRB</b>	APB bridge	Write strobes. This signal indicates which byte lanes to update during a write transfer. There is one write strobe for each eight bits of the write data bus. Therefore, <b>PSTRB[n]</b> corresponds to <b>PWDATA[(8n + 7):(8n)]</b> . Write strobes must not be active during a read transfer.
<b>PREADY</b>	Slave interface	Ready. The slave uses this signal to extend an APB transfer.
<b>PRDATA</b>	Slave interface	Read Data. The selected slave drives this bus during read cycles when <b>PWRITE</b> is LOW. This bus can be up to 32-bits wide.
<b>PSLVERR</b>	Slave interface	This signal indicates a transfer failure. APB peripherals are not required to support the <b>PSLVERR</b> pin. This is true for both existing and new APB peripheral designs. Where a peripheral does not include this pin then the appropriate input to the APB bridge is tied LOW.

Figure 5.11: Table of signals in the APB protocol (Courtesy of ARM [3]).

The protocol contains two individual buses for read and write operations. However, only one of these transfers can be executed at a time. The read and write transfer using the APB protocol with no wait cycles can be seen in figure 5.12.

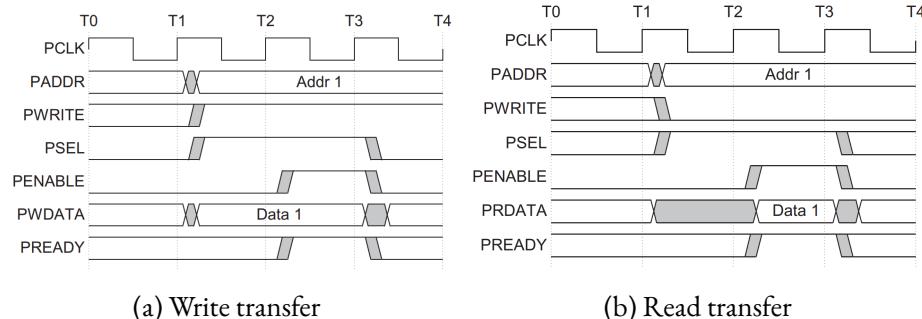


Figure 5.12: Shows the basic transfers of the APB protocol with no wait cycles (Courtesy of ARM [3]).

Wait cycles can be introduced if the slave does not assert the PREADY signal. However, the way the APB protocol is connected to the CPU will cause stalling of the entire CPU until the PREADY is asserted and the transfer is completed. A timeout could partially solve this by limiting the stalling period, but this does not fix the problem but only reduces it. Therefore it is chosen that the IP block as a general rule should always have PREADY asserted, as to ensure no stalling of the system. Instead in the case of a bad transfer due to no data available or similar situations, the PSLVERR signal is utilized and the IP block will then assert this signal to indicate that the transfer has failed. This introduces problems as this signal does not indicate why the transfer failed. However, it does remove the stalling. Therefore, this method is chosen.

The APB bus will not be triplicated as this will connect all peripherals to the core and the length can therefore be quite significant, which will result in a larger area used and more complex routing. Instead, an encoding approach is used. Every byte of the bus is encoded using Hamming codes, which are capable of single error correction and double error detection. This approach uses the same logic developed later for radiation-tolerant memories.

# CHAPTER 6

## ENTER TITLE HERE

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# CHAPTER 7

## CONCLUSION

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## A P P E N D I X A

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