



POLITECNICO MILANO 1863

Reverse Engineering of Juno Mission Homework 7

Course of Space Systems Engineering & Operations
Academic Year 2023-2024

Group 5

Alex Cristian Turcu	alexcristian.turcu@mail.polimi.it	10711624
Chiara Poli	chiara3.poli@mail.polimi.it	10731504
Daniele Paternoster	daniele.paternoster@mail.polimi.it	10836125
Marcello Pareschi	marcello.pareschi@mail.polimi.it	10723712
Paolo Vanelli	paolo.vanelli@mail.polimi.it	10730510
Riccardo Vidari	riccardo.vidari@mail.polimi.it	10711828

Contents

Contents	i
Notation	i
1 Introduction of Juno's configuration	1
2 Shape and appendages	1
3 Configuration inside the launcher	1
4 External configuration	2
4.1 Propulsion subsystem	2
4.2 TMTC subsystem	2
4.3 AOCS subsystem	2
4.4 TCS subsystem	2
4.5 EPS subsystem	3
5 Internal configuration	3
Bibliography	4

Notation

SRU	Stellar Reference Unit	IC	Inner Cruise
SSS	Spinning Sun Sensor	OC	Outer Cruise
JEDI	Jupiter Energetic-particle Detector Instrument	EGA	Earth Gravity Assist
JADE	Jovian Auroral Distribution Experiment	CG	Center of Gravity
REM	Rocket Engine Mount	AOCS	Attitude and Orbit Control System
HGA	High Gain Antenna	TCS	Thermal Control System
MGA	Medium Gain Antenna	TMTC	Telemetry and TeleCommand
LGA	Low Gain Antenna	RCS	Reaction Control System
TLGA	Toroidal Low Gain Antenna	DSM	Deep Space Manuever
JIRAM	Jovian Infra -Red Aurora Mapper	SEP	Sun Earth Probe
UVS	Ultra Violet Spectrograph	EPM	Earth Pointing Mode
MAG	Magnetometer	SPM	Sun Pointing Mode
Ax	Array # x	FOV	Field of View
ME	Main Engine	MLI	Multi-Layer Insulator
		S/C	Spacecraft

1 Introduction of Juno's configuration

Juno is the first spin-stabilized solar-powered spacecraft to conduct science operations around Jupiter, featuring different instruments with various requirements and limitations. Given the harsh environment it faces around the planet, the longevity of the mission and the required power to conduct operations, both packed and unpacked configuration of the payloads, antennas, internal hardware and power sources required special attention. Mass distribution is also a critical factor in the accuracy of the different measurement as Juno spins at 2 RPM during science operations. In the following paragraphs, first the general configuration of Juno will be carried out, then the analysis of the various payloads will be presented.

2 Shape and appendages

The main body of the S/C has the shape of a regular hexagonal prism with sides of 1.77 m and and high of 1.52 m: in its interior the ME, tanks of fuel and oxidizer are stored. The interior of the prism is divided into six bays thanks to honeycomb composite walls that allow, together with MLIs and thermal blanket, to decouple each section from their surroundings. A tank is placed in each bay.

On the top deck are installed the two 55 Ah Li-Ion batteries, the two SRUs, the SSSes, all the cabling for the instruments, JEDI and JADE. Moreover two of the four REMs are placed on the top deck along the Y-axis, the titanium vault is placed on the center of the prism, aligned with the Z-axis. Inside the vault all the important electronics is stored in order to keep it shielded from the radiations. At last, the 2.5 m diameter HGA to transmit and receive in X-band and Ka-band is mounted on top of the vault. Alongside the HGA, the front LGA and the MGA are placed. On three of the six lateral faces of the main body the three solar arrays are mounted and spaced 120° apart. Under the solar array along the +X-axis **REFERENCE** the WAVES antennas are placed, while on a fourth face both the JunoCam and the UVS instruments are mounted. The last two lateral faces are occupied by the Microwaves Radiometer. The three solar arrays are composed by a different number of panels: A1 has three panels while A2 and A3 are composed by 4 panels each. In fact on the edge of A1 the MAG suites of instruments is mounted. The aft deck, aligned with the -Z-axis, features the ME cover, the last two REMs for the ADCS, the aft LGA, the TLGA and JIRAM.

3 Configuration inside the launcher

As mentioned in [section 1](#), the cross section of Juno's main body is an hexagon with 1.77 m sides. The circumscribed circumference has thus a diameter of 3.54 m. The solar arrays have a total area of about 60 m² and their total aperture when deployed is more than 20 m.^[1] Given that the needed launcher, the Atlas V, has as an option a fairing of diameters up to 4.57 m, the three arrays had to be folded. A system of hinges and struts and a division in multiple panels had to be developed.^[2] Dimensions of the arrays led to difficult on ground testing: technicians had to fold and unfold the three arrays one by one and not all three together.^[3] The adopted fairing features a maximum height of 10.18 m, that is the smallest fairing among the 5 m family.^[4] Dimensions are coherent with the ones of Juno. The smallest possible fairing to be mounted on the Atlas V had an internal available diameter of 3.75 m and an available internal height of 1.58 m which was not compliant with dimensions of the S/C. The packed configuration features, in addition to the folded arrays, the Waves antennas to be retracted: together with the solar arrays they will be deployed soon after separation from the Centaur upper stage. The adapter used to connect the launcher to the spacecraft was found to be the type D1666, composed by two pieces of machined aluminum. Loads are transferred from the Centaur to the S/C via its internal panels. Every honeycomb panel is used to divide the 6 bays where the tanks are placed and they all converge in a central column, probably made of titanium, that guarantees internal rigidity. Inside the column the ME is stored. The capabilities of the adapter vary depending on the position of the CG of the spacecraft with respect to the separation plane. Due to the mass distribution of Juno at launch, the CG was found to be inside the main body, at around 1.4 m from the said plane. From the Atlas V user manual^[4], this values imply a maximum payload capability of 6.5 tons, above the ≈ 3.62 tons of Juno at launch.^[1] In [Figure 1](#)^[5] it is possible to observe the S/C with the folded arrays and the adapter prior to the launch. Attention must be paid to A1, where the MAG is mounted. Separation of Juno from the adapter is carried out by a system of springs, a clamp band and a release mechanism: the system allows to safely separate the S/C and provides the needed energy to obtain positive separation, which in the case of Juno implies reaching an orbit with a specific energy of 31.1 km²/s².



Figure 1: Juno in packed configuration

4 External configuration

Juno is a spin stabilized S/C that performs science operations in a LILT environment, with a trajectory ranging between 0.85 AU and 5.45 AU. The positioning of the instruments and the scientific payload is defined such that during science operations each face of the main body has a clear view of Jupiter twice a minute. ICs and OC are performed at 1 RPM while all the critical maneuvers are conducted at 2 RPM. All the on board subsystems will be analyzed.

4.1 Propulsion subsystem

RCS: the twelve RCS thrusters are mounted on four REMs, three thrusters each, two REMs are on the forward deck (F-REMs) and two are on the aft deck (A-REMs). Each REM is mounted on top of a pylon: F-REMs are raised by 74 cm while A-REMs are raised by 26 cm from their respective decks. Each thruster features a different orientation with respect to the Z and X axis: axial thrusters are canted 10° from the Z-axis on the top deck. Lateral thrusters are canted 5° from the X-axis and 12.5° from the Z-axis. Axial thrusters can be utilized as an alternative to the main engine. **REFERENCE** All this precautions are needed as plum from the firing of the thruster must not interact nor with the solar arrays nor with the antennas.

ME: the Leros 1b main engine is stored inside the main body. On the aft deck its cover is visible: this device is needed to protect the engine from collisions with debris along the cruise.

4.2 TMTC subsystem

HGA: this antenna is both responsible for communications along the mission and for conducting science operations. It is mounted on top of the electronic vault and has a diameter of 2.5 m. It is located along the spin axis, such that its pointing requirements of 0.25° are satisfied. In addition to transmitting and receiving in X or Ka-band, the HGA also serves the fundamental role of thermally protecting the spacecraft from the radiations coming the Sun during the IC: the whole antenna is covered with a highly reflective Germanium Kapton blanket.

MGA: this antenna is mounted near the HGA on the forward deck and it is used for communications only. It serves as alternative to the HGA during the different maneuvers as its beamwidth is larger at 10.3° .

LGA: two of this type of antennas are present on Juno, on the forward deck, on the same structure of the MGA, and one on the aft deck. The two antennas are coupled together as are not independently controllable and are only used during the EGA due their limited power.

TLGA: this antenna serves a crucial role during all the maneuvers that Juno performs and in the event of a safe mode. It is mounted on the aft deck, aligned with A1, and has the peculiarity to be a biconical horn style antenna. The TLGA transmits only tones that are necessary to assess the health status of the spacecraft. During the DSMs the SEP angle must be carefully monitored in order to not exceed safe operational values for this antenna.

4.3 AOCS subsystem

The **RCS** used to control the attitude of the S/C is presented in [subsection 4.1](#)

SRU: two of this type of sensor are mounted on the forward deck of Juno, looking in the radial direction. They are responsible for the attitude determination of the S/C during all the ICs, the OC, and science operations. Software calculations are required as the spin rate of Juno changes throughout the mission, from 1 to 2 RPM. The SRUs are turned off and covered during the two DSMs as a direct sight of the Sun would have damaged their sensible optics. While performing EPM or SPM no precautions are needed.

SSS: two of Spinning Sun Sensors are positioned on the edge of the forward deck and are oriented in such a way to include both the Z-axis and a portion of the XY plane inside their FOV. They are used during critical maneuvers to provide accurate attitude determination while the SRUs are not in working conditions and allow for a fail safe recovery.

4.4 TCS subsystem

MLI: this kind of passive thermal control is needed as the trajectory of Juno ranges between 0.88 AU and 5.45 AU. Active thermal control is not the only option as it requires power, a precious resource around Jupiter. Moreover, MLIs are also used to shield the body and the instruments from the radiation present around Jupiter.

Solar Arrays: this massive structure is composed by eleven panels of different sizes. The panels are composed by a carbon fiber support, on which solar cells are placed. The cells are placed on a Kapton film and protected by an AR glass. The backside of the panels is covered in Black Kapton. This is done to ensure that the operability range of temperatures of the cells is met across the different phases of the mission, from the $+90^\circ\text{C}$ at the perihelion to the -146°C around Jupiter. Moreover, the panels have to dissipate the excess of energy, stored as heat, that they produce during the first phases of the mission. On A1 the MAG boom is mounted: its instruments require special attention as a strong temperature gradient, $\approx 80^\circ\text{C}$, is present. The MAG boom and the MOBs are thus thermally decoupled via titanium made hinges and joints.

Vault: this particular element, made of titanium, is positioned on the centre of the forward deck of the S/C and its dimensions are $\approx 0.8 \text{ m} \times 0.8 \text{ m} \times 0.7 \text{ m}$. It houses all the electronics as it is able to shield it from the radiations around Jupiter. It also succeeds in keeping its interior inside the correct range of temperature thanks to the presence of heaters and louvers. The position of the louvers is critical as they must radiate to the deep space and not face other instruments on the deck: thermal control prior the EGA and during science operation is critical as any loss inside the vault could lead to the loss of the mission. During operations it is thermally shielded by the HGA.

Forward deck instruments:

4.5 EPS subsystem

MLI:

Solar Arrays:

Vault:

Forward deck instruments:

5 Internal configuration

Bibliography

- [1] NASA JPL. *Juno Quick Facts*. Site: https://www.jpl.nasa.gov/news/press_kits/juno/facts/. 2011.
- [2] William McAlpine et al. "JUNO Photovoltaic Power at Jupiter". In: (2012).
- [3] Site: <https://www.jpl.nasa.gov/news/juno-solar-panels-complete-testing>.
- [4] Site: <https://www.ulalaunch.com/docs/default-source/rockets/atlasvusersguide2010.pdf>.
- [5] Site: <https://spaceflightnow.com/atlas/av029/fairing/>.