

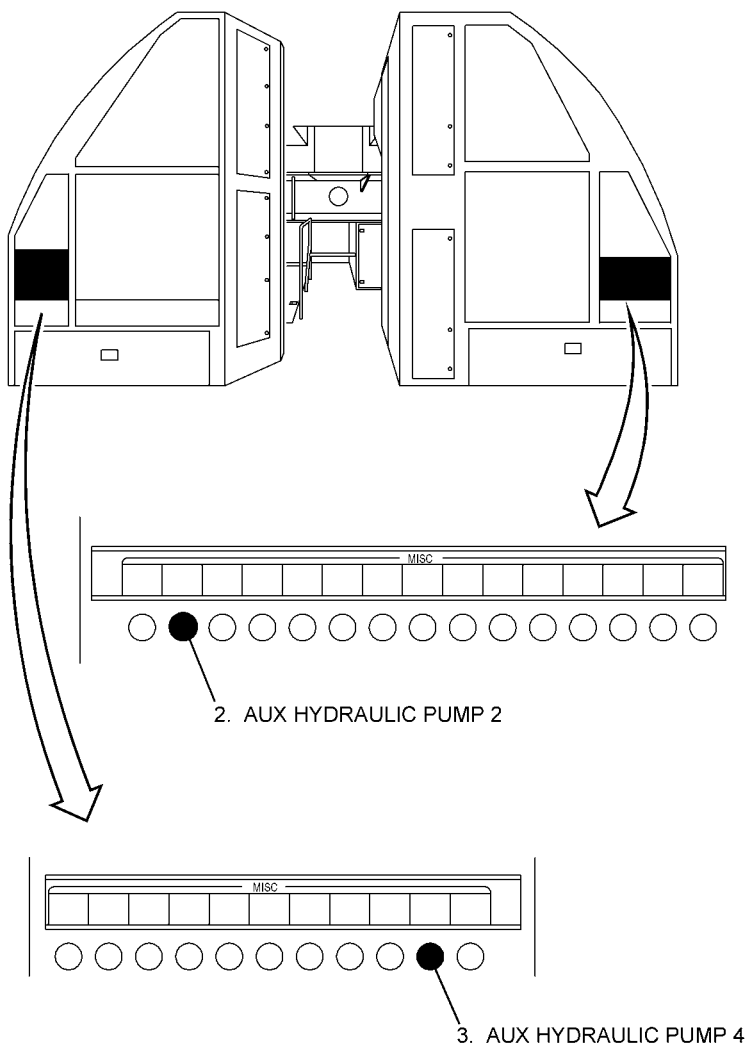
ICN-88277-G2721065-002-01

3-2. FOLLOW-ON MAINTENANCE.

1. Install horizontal pressure panel access cover assembly (53-12-10).

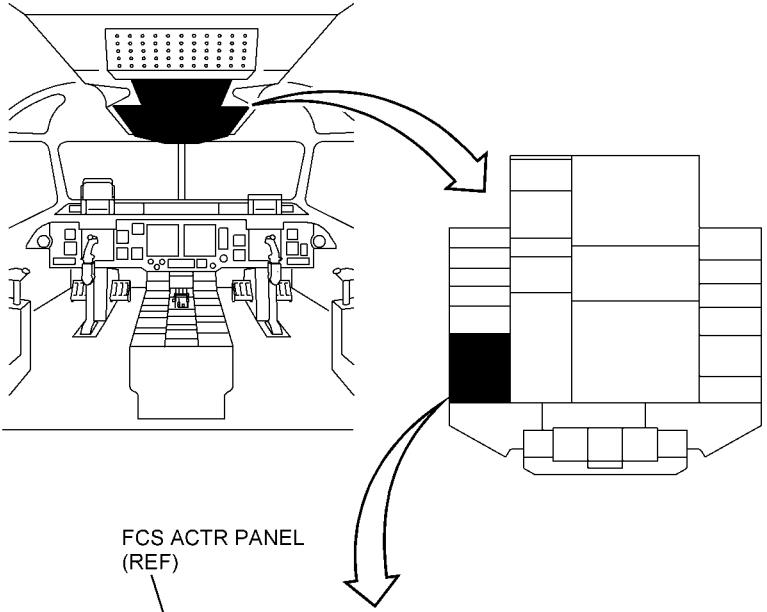
SUPPORT REF DES	PANEL NO.	PANEL REF DES
2721AA009 2721AA010	112AZP	5312CA001
2721AA011 2721AA012	112BZP	5312CA002

2. (B) Remove warning tag and close **AUX HYDRAULIC PUMP 2** circuit breaker on Electrical Power Center (EPC) row **LL**, column **69**.
3. (B) Remove warning tag and close **AUX HYDRAULIC PUMP 4** circuit breaker on EPC row **LL**, column **10**.

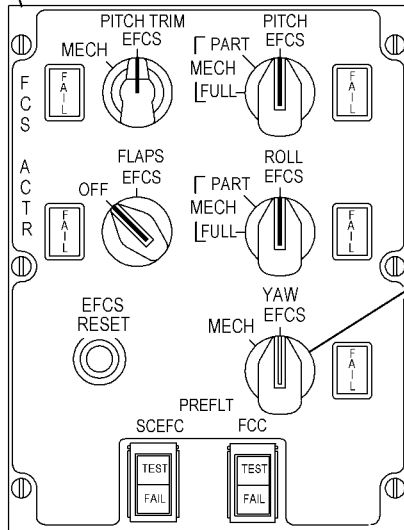


ICN-88277-G2721068-003-01

4. (B) Remove warning tag from **YAW** switch, on **FCS ACTR** panel.



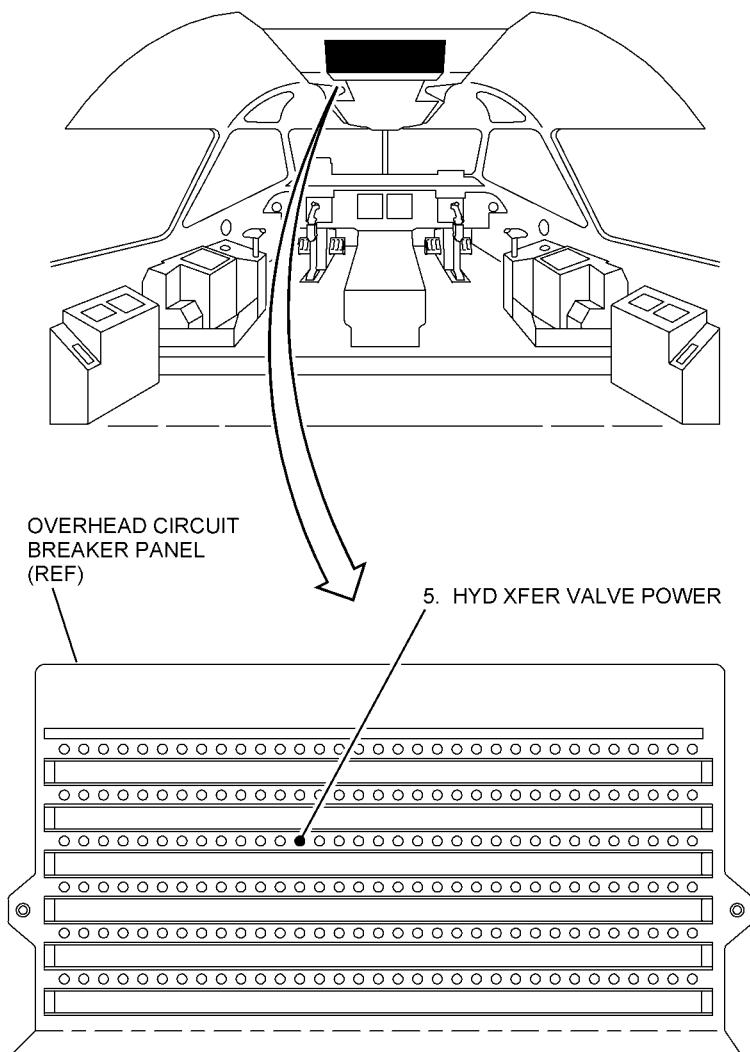
FCS ACTR PANEL
(REF)



4. YAW

ICN-88277-G2721364-003-01

5. (B) Remove warning tag and close **HYD XFER VALVE POWER** circuit breaker on overhead panel, row **C**, column **13**.

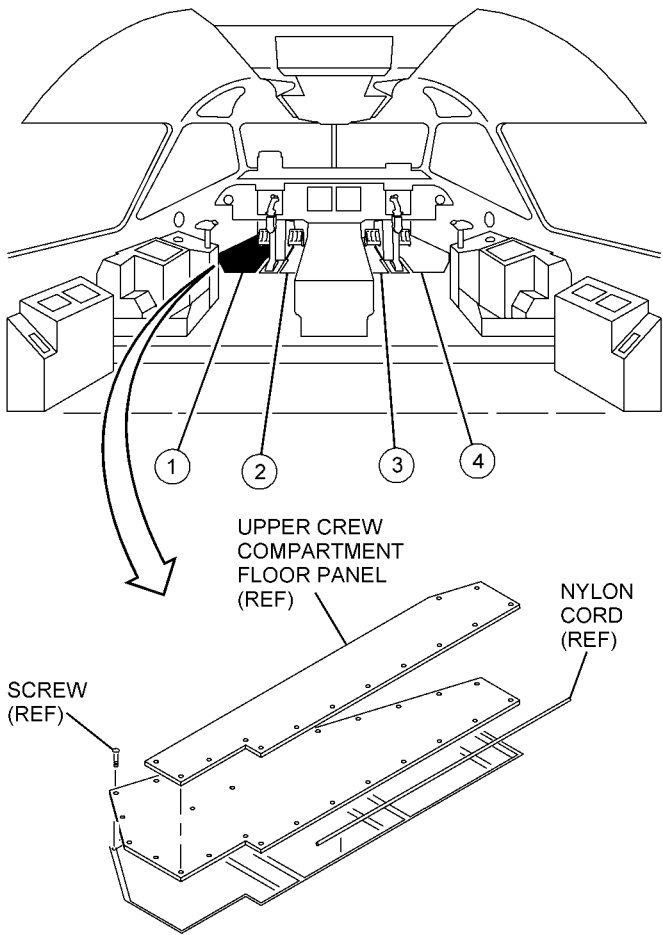


ICN-88277-G2721070-004-01

TO 1300i-2-27JG-20-2

- 6. Perform tape type sealing (TO 1300i-23, Chapter 1, Section III).
- 7. (A) Position upper crew compartment floor panels.
- 8. Perform fastener head sealing (TO 1300i-23, Chapter 1, Section III).
- 9. (A) Install screws as identified.
- 10. Perform floor panel sealing (TO 1300i-23, Chapter 1, Section III).
- 11. Install pilot/copilot aircraft seat (25-11-10, task 3-1) as follows:

SUPPORT REF DES	SEAT REF DES
2721AA009 2721AA010	2511AA001
2721AA011 2721AA012	2511AA002



7. |
9. |

SUPPORT REF DES	INDEX NO	PANEL NO	SCREW QTY	NYLON CORD QTY
2721AA009	1	211AZY	28	7
2721AA010	2	211BZY	21	6
2721AA011	3	211CZY	21	6
2721AA012	4	211DZY	28	7

(TYPICAL)

ICN-88277-G2721463-002-01

CARTRIDGE-RUDDER AND ELEVATOR FORCE LIMITER SPRING ASSEMBLY (27-21-13)

MASTER INPUT CONDITIONS:

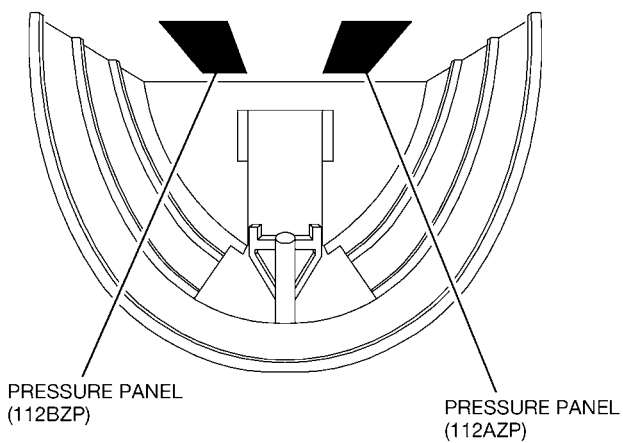
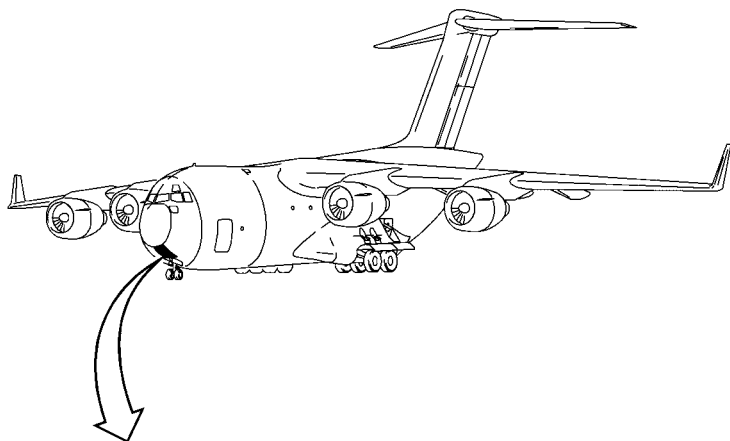
Reference designators:

2721AA023	Cartridge-Rudder and Elevator Force Limiter Spring Assembly
-----------	--

Applicable functions:

- 2 Removal.
- 3 Installation.

Access data:



ICN-88277-G2721071-002-01

CARTRIDGE-RUDDER AND ELEVATOR FORCE LIMITER SPRING ASSEMBLY REMOVAL (27-21-13-2)

FUNCTIONAL INPUT CONDITIONS:

Applicability:

	Task
All	All

Additional information:

This procedure consists of the following tasks:

- 2-1. Preparation.
- 2-2. Removal.

Additional data:

	Task
TO 1300i-2-05JG-10-1	2-1
TO 1300i-2-53JG-10-1	2-1

Personnel recommended:

	Task
One	All

Safety conditions:

Task

WARNING

The horizontal pressure panel access cover(s) are removed in these tasks to gain access to the cavity above. When rudder, aileron, and elevator aircraft ground safety locks are not installed, care shall be taken working around rudder, aileron, and elevator cables, pulleys, and linkage due to possible moving parts. Failure to comply could result in injury to personnel.

All

Support equipment:

<u>Nomenclature</u>	<u>PN</u>	<u>Specification</u>	<u>Qty</u>	<u>Task</u>
Kit, Pin Rig	17G140185-1	--	--	--
Pin 5-12, Rig	17G140185-5	--	1	2-1

Supplies:

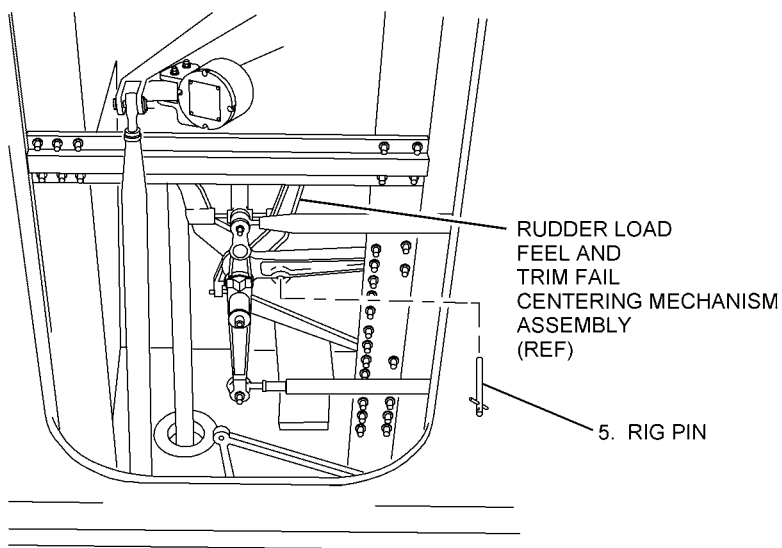
<u>Nomenclature</u>	<u>PN</u>	<u>Specification</u>	<u>Qty</u>	<u>Task</u>
NA	--	--	--	--

2-1. PREPARATION.

- 1. Review "Section 1 (General Information)" of this TO for system general warnings, cautions, and notes.
- 2. Review task "Functional Input Conditions" page for task specific safety conditions.
- 3. Install rudder aircraft ground safety lock (05-10-01, task 01-1).
- 4. Remove horizontal pressure panel access cover assemblies (53-12-10).

PANEL NO.	PANEL REF DES
112AZP	5312CA001
112BZP	5312CA002

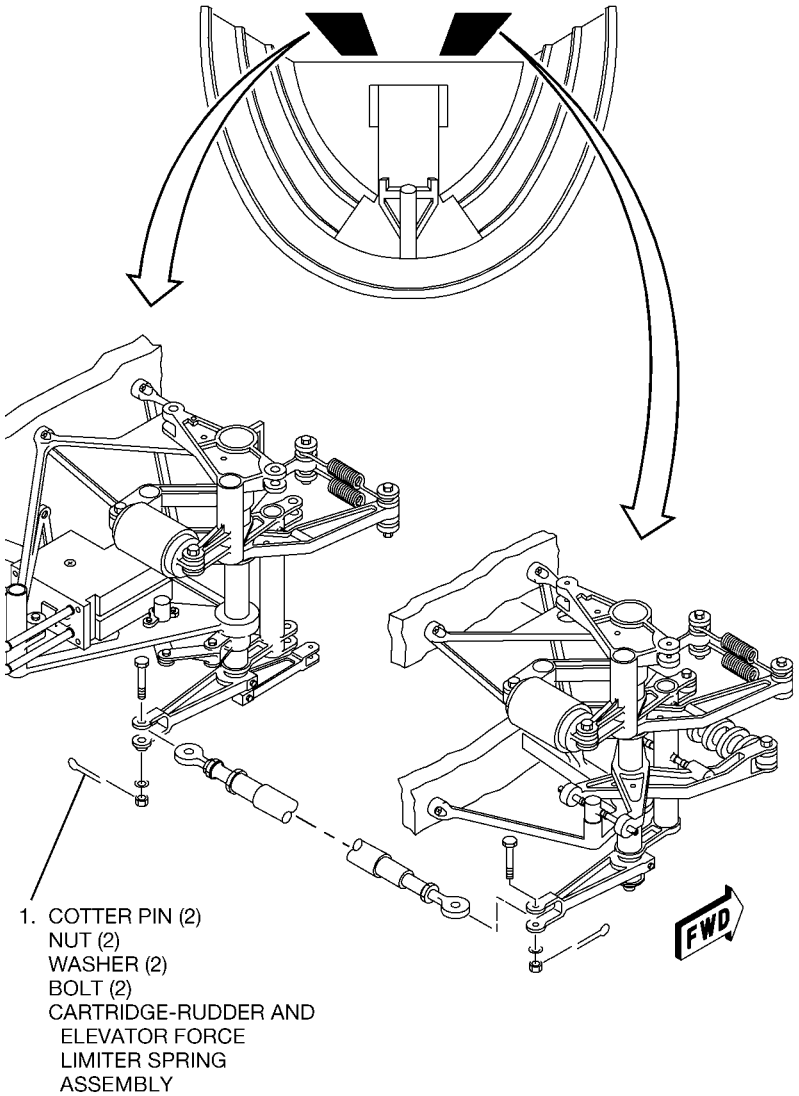
- 5. Install rig pin 5-12 in rudder load feel and trim fail centering mechanism assembly.



ICN-88277-G2721073-004-01

2-2. REMOVAL.

1. Remove cotter pins, nuts, washers, bolts, and cartridge-rudder and elevator force limiter spring assembly.



ICN-88277-G2721074-003-01

CARTRIDGE-RUDDER AND ELEVATOR FORCE LIMITER SPRING ASSEMBLY INSTALLATION (27-21-13-3)

FUNCTIONAL INPUT CONDITIONS:

Applicability:

All

Task

All

Additional information:

This procedure consists of the following tasks:

- 3-1. Installation.
- 3-2. Follow-on maintenance.

Additional data:

TO 1300i-2-05JG-10-1

TO 1300i-2-27JG-20-1

TO 1300i-2-53JG-10-1

Task

3-2

3-2

3-2

Personnel recommended:

One

Task

All

Safety conditions:

Task

WARNING

The horizontal pressure panel access cover(s) are removed in these tasks to gain access to the cavity above. When rudder, aileron, and elevator aircraft ground safety locks are not installed, care shall be taken working around rudder, aileron, and elevator cables, pulleys, and linkage due to possible moving parts. Failure to comply could result in injury to personnel.

All

Support equipment:

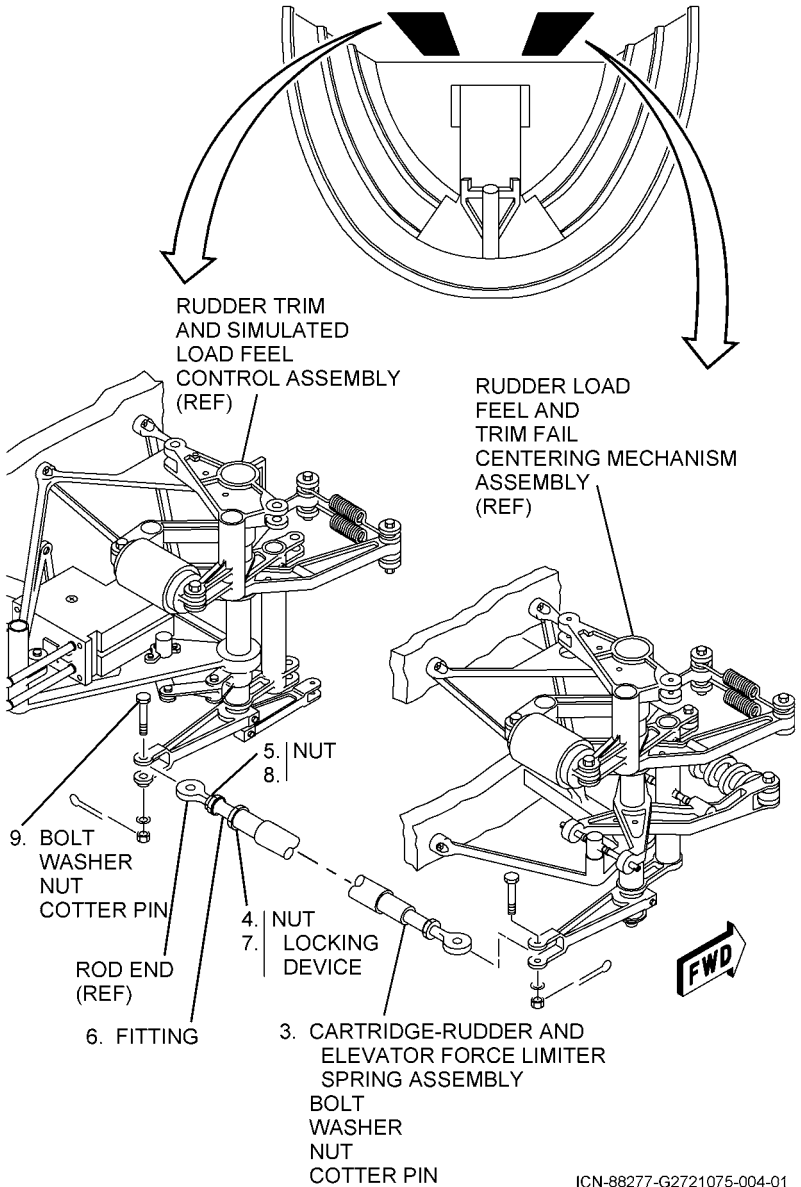
<u>Nomenclature</u>	<u>PN</u>	<u>Specification</u>	<u>Qty</u>	<u>Task</u>
NA	--	--	--	--

Supplies:

<u>Nomenclature</u>	<u>PN</u>	<u>Specification</u>	<u>Qty</u>	<u>Task</u>
Pin, Cotter	MS24665-151	--	2	3-1
Wire, Safety	900010-32C	--	AR	3-1

3-1. INSTALLATION.

1. Review "Section 1 (General Information)" of this TO for system general warnings, cautions, and notes.
2. Review task "Functional Input Conditions" page for task specific safety conditions.
3. Position cartridge-rudder and elevator force limiter spring assembly in rudder load feel and trim fail centering mechanism assembly; install bolt, washer, nut, and cotter pin.
4. Remove safety wire; loosen nut locking device.
5. Loosen nut on rod end.
6. Adjust fitting for installation.
7. Position nut locking device and tighten; secure with safety wire.
8. Tighten nut.
9. Install bolt, washer, nut, and cotter pin.



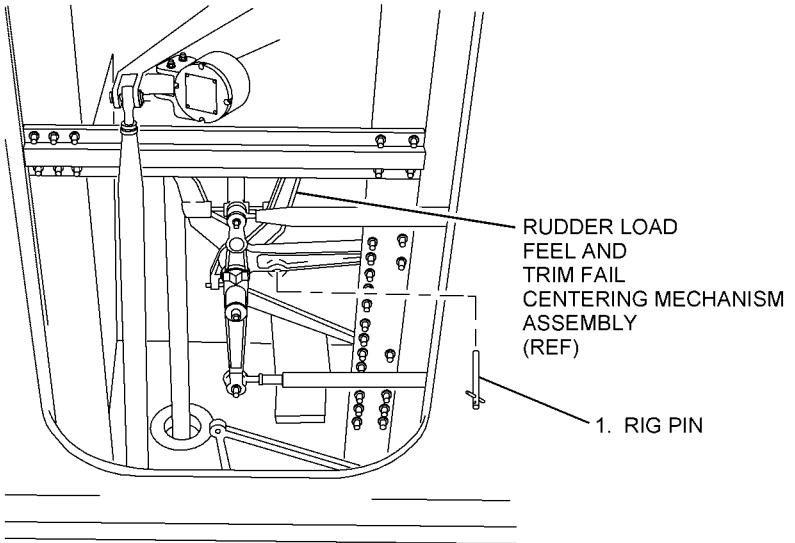
ICN-88277-G2721075-004-01

3-2. FOLLOW-ON MAINTENANCE.

- 1. Remove rig pin from rudder load feel and trim fail centering mechanism assembly.
- 2. Install horizontal pressure panel access cover assemblies (53-12-10).

PANEL NO.	PANEL REF DES
112AZP	5312CA001
112BZP	5312CA002

- 3. Remove rudder aircraft ground safety lock (05-10-01, task 01-2).
- 4. Perform rudder system operational checkout (27-20-01).



ICN-88277-G2721076-004-01

RUDDER PEDAL STATURE ADJUST CRANK ASSEMBLY (27-21-14)

MASTER INPUT CONDITIONS:

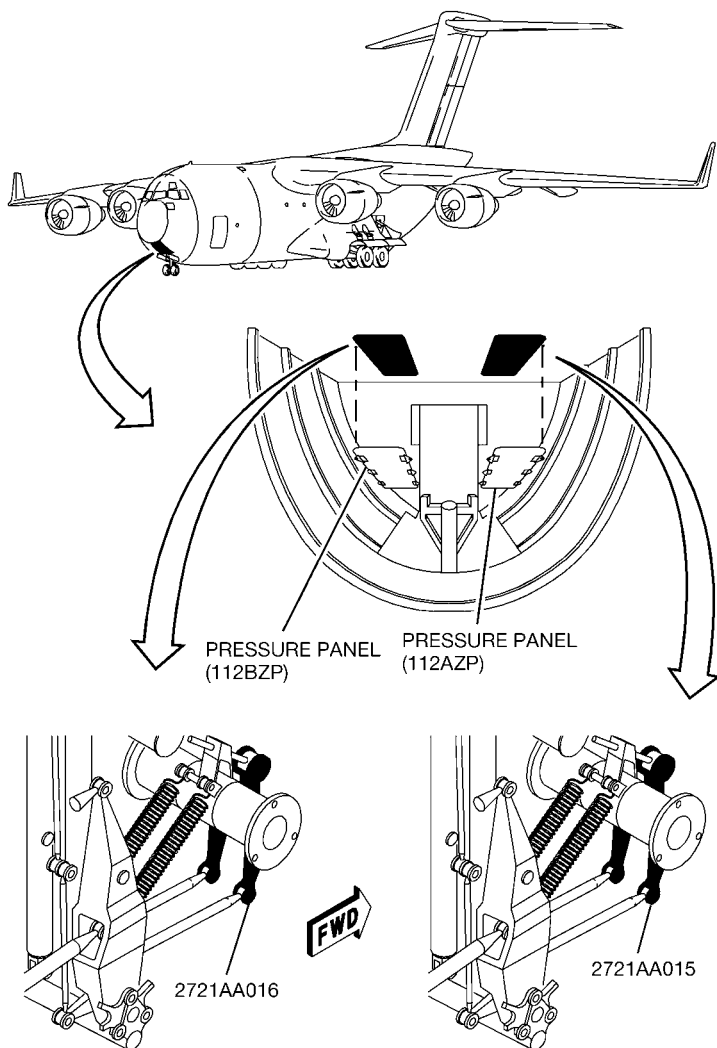
Reference designators:

2721AA015	Rudder Pedal Stature Adjust Crank Assembly
2721AA016	Rudder Pedal Stature Adjust Crank Assembly

Applicable functions:

- 2 Removal.
- 3 Installation.
- 4 Repair.

Access data:



ICN-88277-G2721091-003-01

RUDDER PEDAL STATURE ADJUST CRANK ASSEMBLY REMOVAL (27-21-14-2)

FUNCTIONAL INPUT CONDITIONS:

Applicability:

All

Task

All

Additional information:

This procedure consists of the following tasks:

- 2-1. Preparation.
- 2-2. Removal.
- 2-3. Follow-on maintenance.

NOTE

This is typical removal task for all rudder pedal stature
adjust crank assemblies.

Task

All

Additional data:

TO 1300i-2-05JG-10-1

TO 1300i-2-53JG-10-1

Task

2-1

2-1

Personnel recommended:

One

Two

Person (A) performs task.

Person (B) assists person (A).

Task

2-1,
2-3

2-2

Safety conditions:

Task

WARNING

The horizontal pressure panel access cover(s) are removed in these tasks to gain access to the cavity above. When rudder, aileron, and elevator aircraft ground safety locks are not installed, care shall be taken working around rudder, aileron, and elevator cables, pulleys, and linkage due to possible moving parts. Failure to comply could result in injury to personnel.

All

Support equipment:

<u>Nomenclature</u>	<u>PN</u>	<u>Specification</u>	<u>Qty</u>	<u>Task</u>
Tool, Turnbuckle Adjustment	17G140019-1	--	1	2-2

Supplies:

<u>Nomenclature</u>	<u>PN</u>	<u>Specification</u>	<u>Qty</u>	<u>Task</u>
NA	--	--	--	--

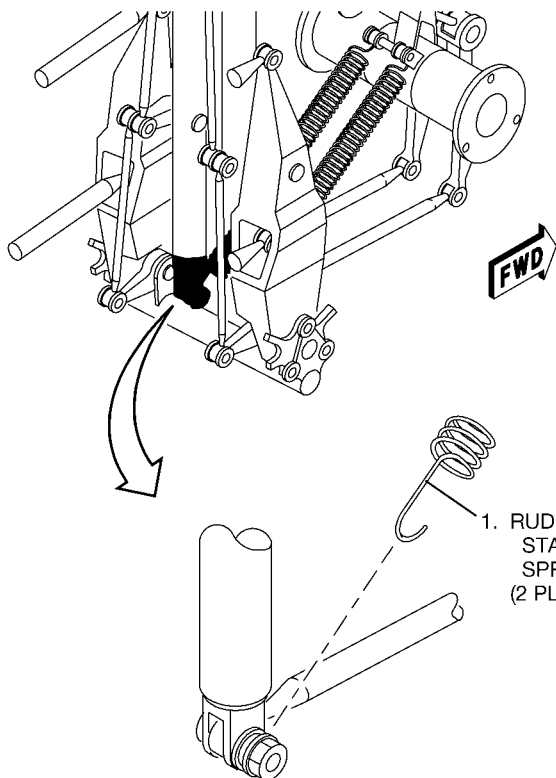
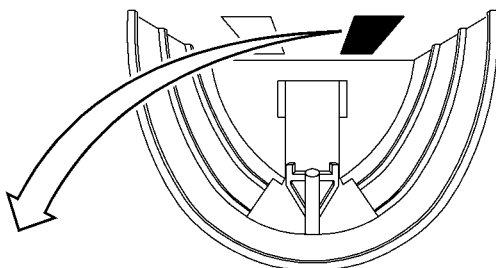
2-1. PREPARATION.

- 1. Review "Section 1 (General Information)" of this TO for system general warnings, cautions, and notes.
- 2. Review task "Functional Input Conditions" page for task specific safety conditions.
- 3. Install rudder aircraft ground safety lock (05-10-01, task 01-1).
- 4. Remove horizontal pressure panel access cover assembly (53-12-10).

CRANK REF DES	PANEL NO.	PANEL REF DES
2721AA015	112AZP	5312CA001
2721AA016	112BZP	5312CA002

2-2. REMOVAL.

1. (A) Disconnect rudder pedal stature return springs.

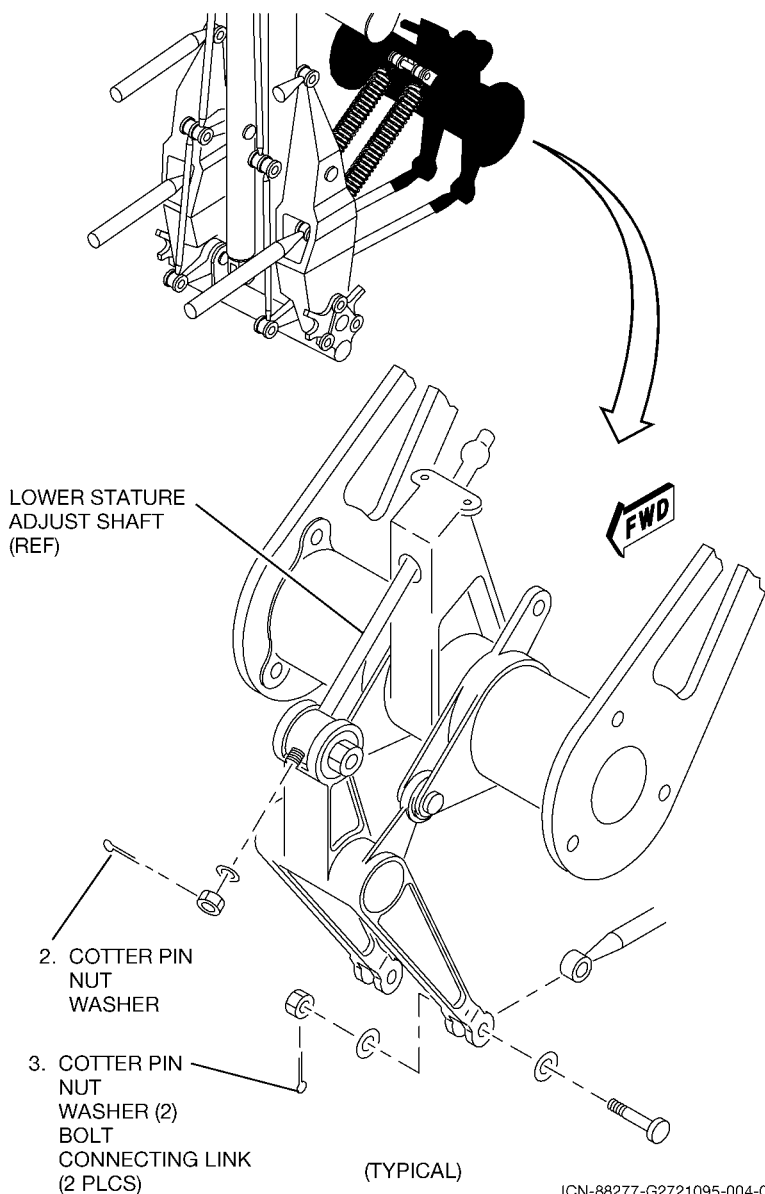


(TYPICAL)

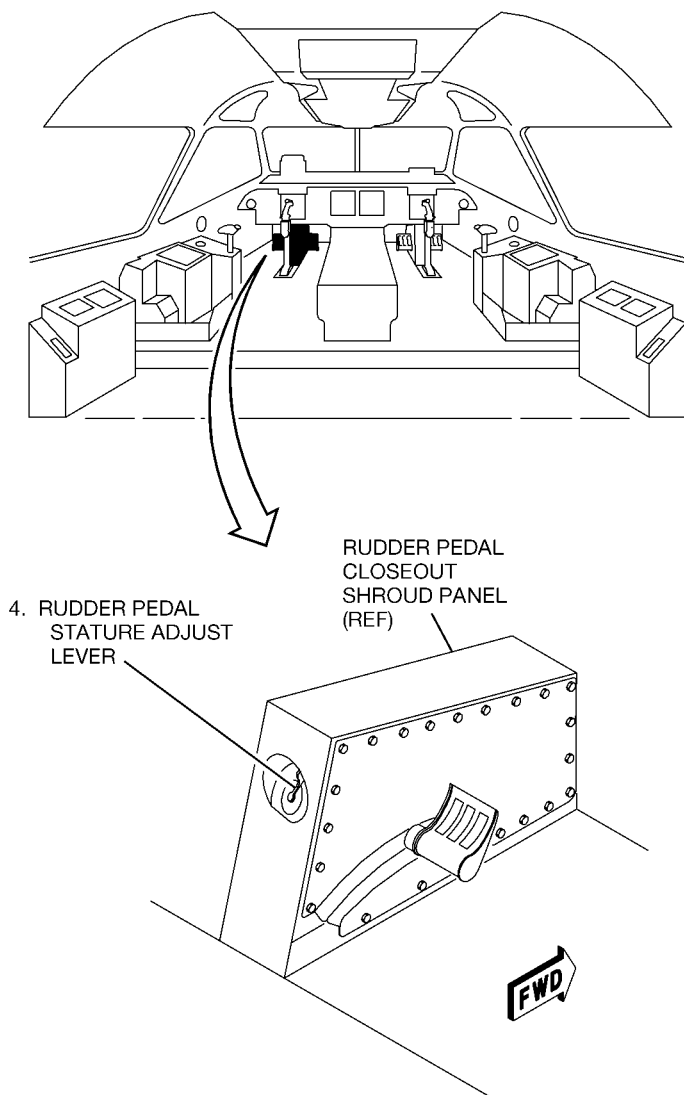
ICN-88277-G2721094-004-01

TO 1300i-2-27JG-20-2

2. (A) Remove cotter pin, nut, and washer from lower stature adjust shaft.
3. (A) Remove cotter pins, nuts, washers, bolts and disconnect connecting links.



4. (B) Rotate rudder pedal stature adjust lever on rudder pedal closeout shroud panel fully clockwise.



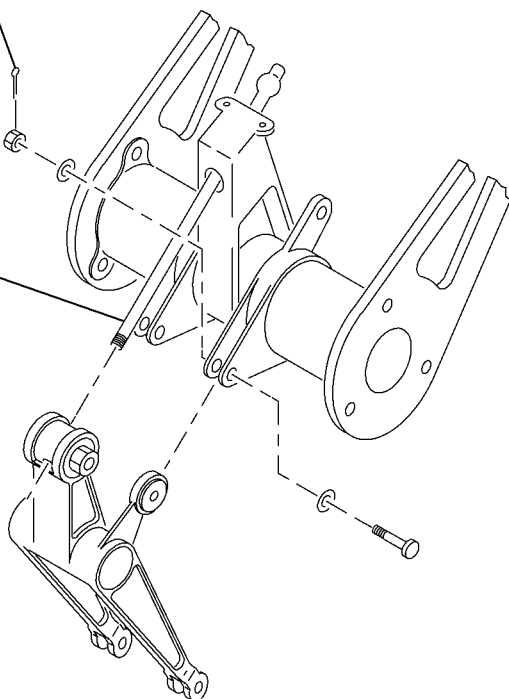
(TYPICAL)

ICN-88277-G2721354-003-01

5. (A) Remove cotter pins, nuts, washers, bolts, and rudder pedal stature adjust crank assembly.

5. COTTER PIN (2)
 NUT (2)
 WASHER (4)
 BOLT (2)
 RUDDER PEDAL
 STATURE ADJUST
 CRANK ASSEMBLY

LOWER STATURE
 ADJUST SHAFT
 (REF)

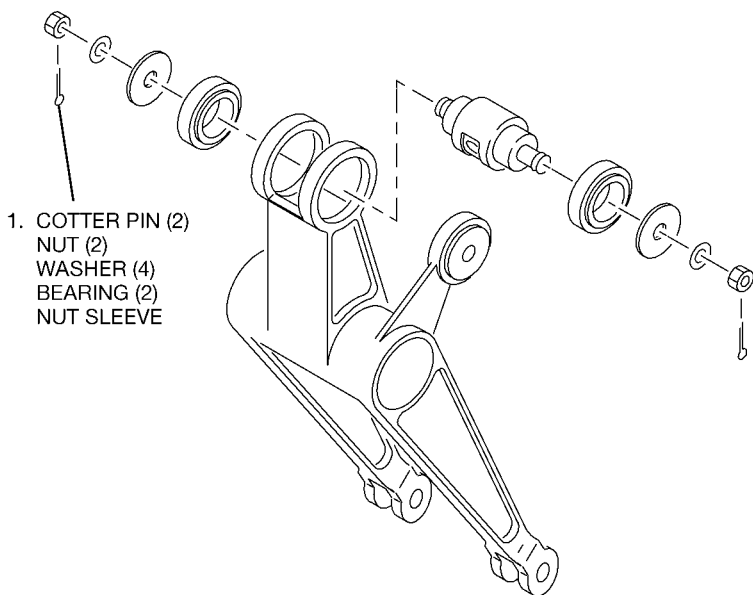


(TYPICAL)

ICN-88277-G2721355-003-01

2-3. FOLLOW-ON MAINTENANCE.

1. Remove cotter pins; identify and remove nuts, washers, bearings and nut sleeve.



(TYPICAL)

ICN-88277-G2721096-002-01

**RUDDER PEDAL STATURE ADJUST CRANK
ASSEMBLY INSTALLATION
(27-21-14-3)**

FUNCTIONAL INPUT CONDITIONS:

Applicability:	Task
All	All

Additional information:

This procedure consists of the following tasks:

- 3-1. Preparation.
- 3-2. Installation.
- 3-3. Follow-on maintenance.

NOTE	Task
This is typical installation task for all rudder pedal stature adjust crank assemblies.	All

Additional data:	Task
TO 1300i-2-05JG-10-1	3-3
TO 1300i-2-27JG-20-1	3-3
TO 1300i-2-53JG-10-1	3-3

Personnel recommended:	Task
One	3-1, 3-3
Two	3-2
Person (A) performs task.	
Person (B) assists person (A).	

Safety conditions:

Task

WARNING

The horizontal pressure panel access cover(s) are removed in these tasks to gain access to the cavity above. When rudder, aileron, and elevator aircraft ground safety locks are not installed, care shall be taken working around rudder, aileron, and elevator cables, pulleys, and linkage due to possible moving parts. Failure to comply could result in injury to personnel.

All

Support equipment:

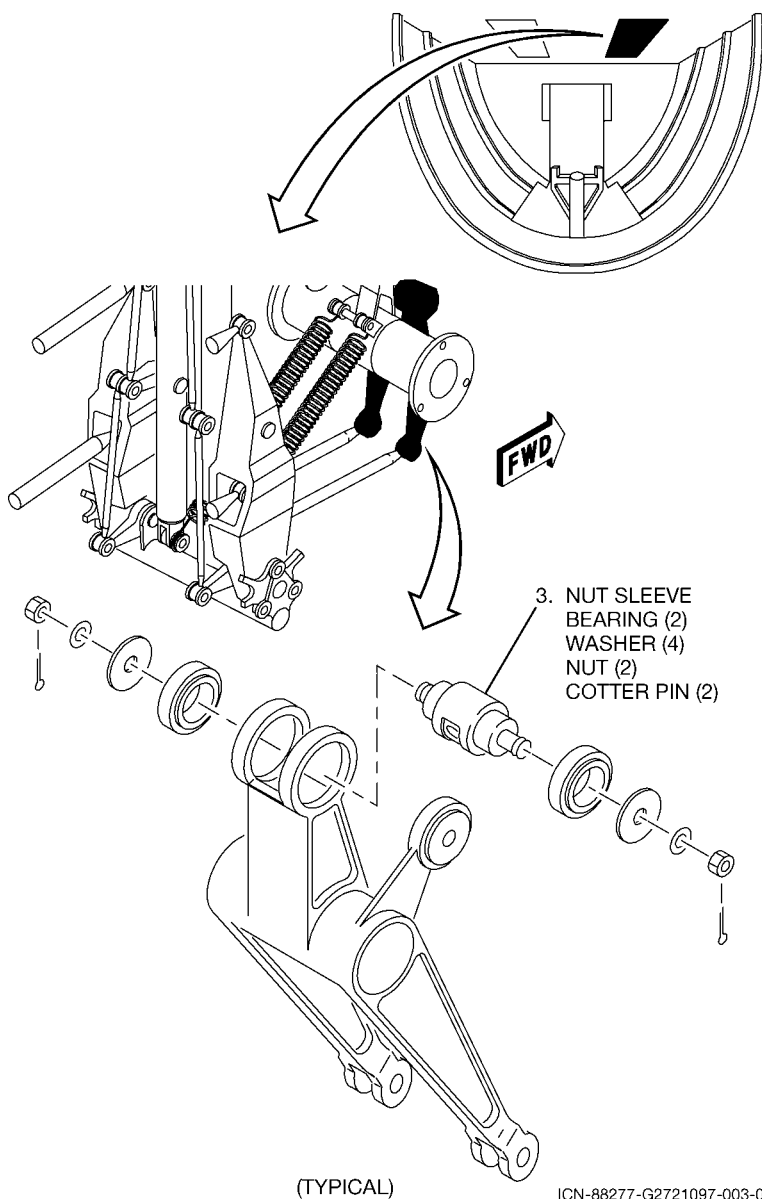
<u>Nomenclature</u>	<u>PN</u>	<u>Specification</u>	<u>Qty</u>	<u>Task</u>
Tool, Turnbuckle Adjustment	17G140019-1	--	1	3-2

Supplies:

<u>Nomenclature</u>	<u>PN</u>	<u>Specification</u>	<u>Qty</u>	<u>Task</u>
Pin, Cotter	MS24665-151	--	5	3-2
Pin, Cotter	MS24665-153	--	2	3-1

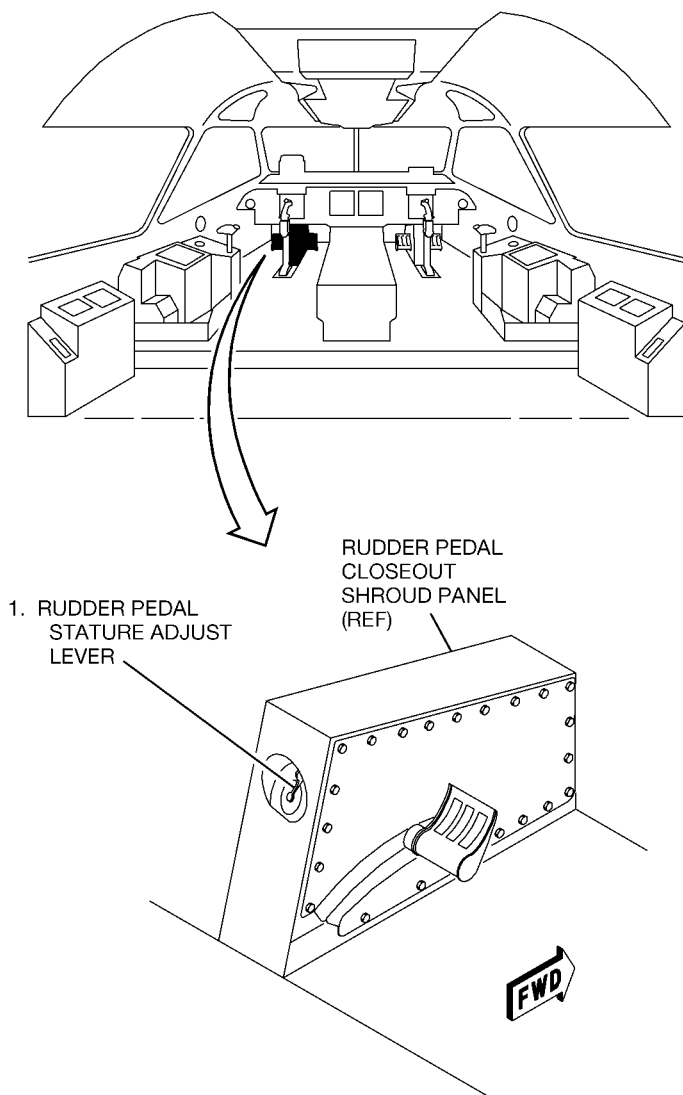
3-1. PREPARATION.

1. Review "Section 1 (General Information)" of this TO for system general warnings, cautions, and notes.
2. Review task "Functional Input Conditions" page for task specific safety conditions.
3. Position nut sleeve and install bearings, washers, and nuts as identified and install cotter pins.



3-2. INSTALLATION.

1. (B) Rotate rudder pedal stature adjust lever on rudder pedal closeout shroud panel fully clockwise.

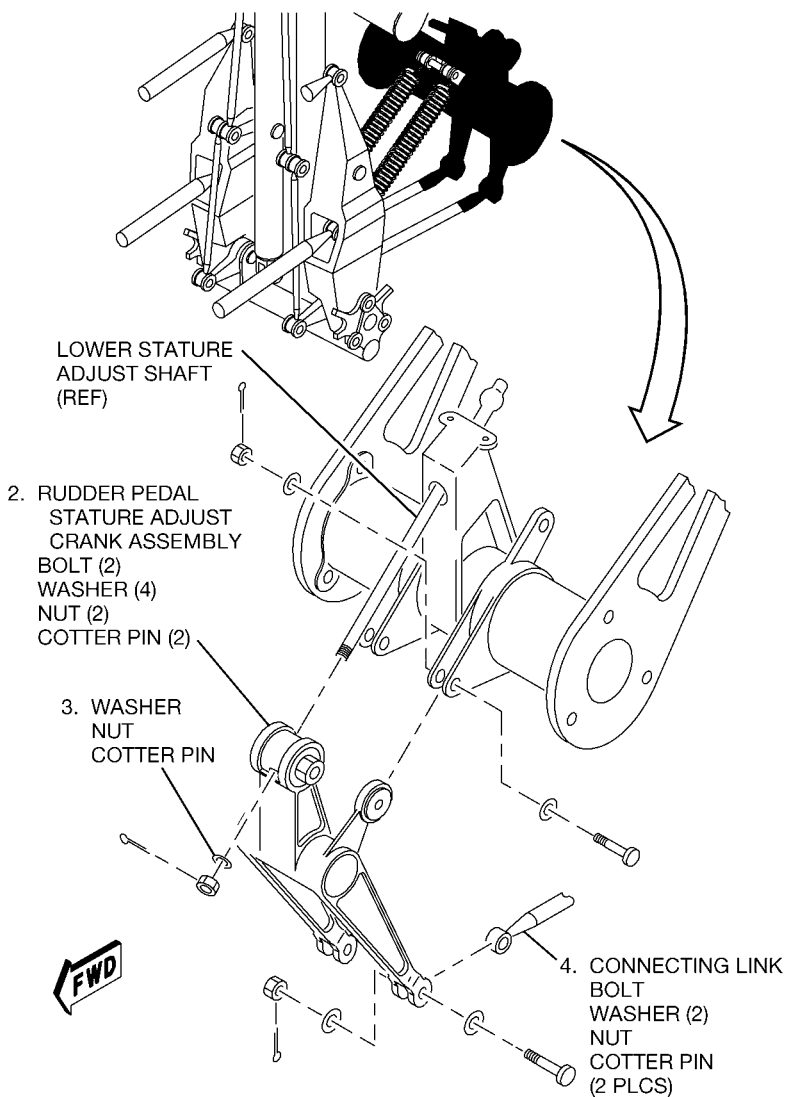


(TYPICAL)

ICN-88277-G2721356-002-01

TO 1300i-2-27JG-20-2

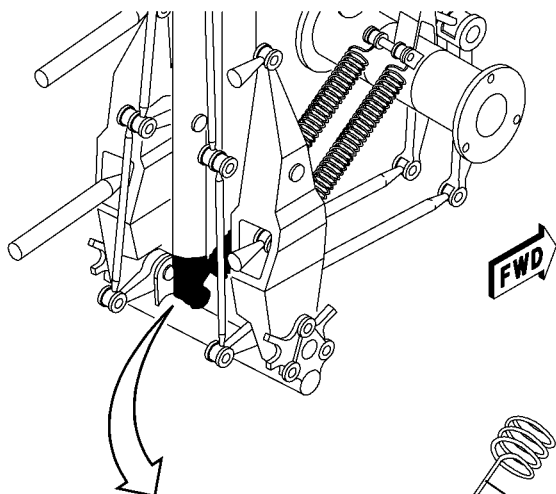
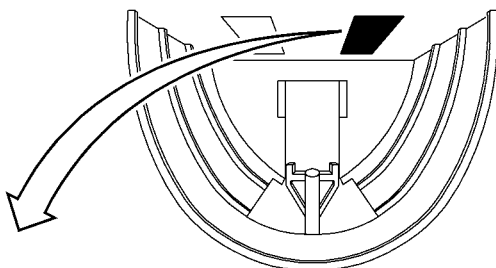
2. (A) Position rudder pedal stature adjust crank assembly and install bolts, washers, nuts, and cotter pins.
3. (A) Install washer, nut, and cotter pin on lower stature adjust shaft.
4. (A) Position connecting links and install bolts, washers, nuts, and cotter pins.



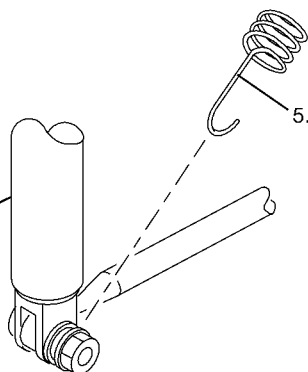
(TYPICAL)

ICN-88277-G2721098-004-01

5. (A) Connect rudder pedal stature return springs on direction control pedal support assemblies.



DIRECTION
CONTROL
PEDAL SUPPORT
ASSEMBLY
(REF)



5. RUDDER PEDAL
STATURE RETURN
SPRING
(2 PLCS)

(TYPICAL)

ICN-88277-G2721099-003-01

3-3. FOLLOW-ON MAINTENANCE.

- 1. Install horizontal pressure panel access cover assembly (53-12-10).

CRANK REF DES	PANEL NO.	PANEL REF DES
2721AA015	112AZP	5312CA001
2721AA016	112BZP	5312CA002

- 2. Perform rudder system operational checkout (27-20-01).
- 3. Remove rudder aircraft ground safety lock (05-10-01, task 01-2).

**RUDDER PEDAL STATURE ADJUST CRANK
ASSEMBLY
(27-21-14-4)**

FUNCTIONAL INPUT CONDITIONS:

Applicability:	Task
All	All

Additional information:

This procedure consists of the following tasks:

- 4-1. Preparation.
- 4-2. Repair rudder pedal stature adjust crank assembly by replacing universal joint.
- 4-3. Repair rudder pedal stature adjust crank assembly by replacing adjustment lever and shaft.
- 4-4. Follow-on maintenance.

NOTE	Task
-------------	-------------

This is a typical repair task for all rudder pedal stature adjustment crank assembly adjustment lever and shaft.	All
--	-----

Additional data:	Task
TO 1300i-2-05JG-10-1	4-1, 4-4
TO 1300i-2-23JG-40-1	4-3
TO 1300i-2-27JG-20-1	4-4
TO 1300i-23	4-2, 4-3
TO 1300i-2-53JG-10-1	4-1, 4-4

Personnel recommended:

Task

One

4-1,
4-4

Two

4-2,
4-3

Person (A) performs task.
Person (B) assists person (A).

Safety conditions:

Task



- Areas below lower wing fixed trailing edge door assembly shall remain clear of personnel when door is in lowered position and suspended from lanyards. Failure to comply may cause injury to personnel or damage to aircraft. All
- The horizontal pressure panel access cover(s) are removed in these tasks to gain access to the cavity above. When rudder, aileron, and elevator aircraft ground safety locks are not installed, care shall be taken working around rudder, aileron, and elevator cables, pulleys, and linkage due to possible moving parts. Failure to comply may cause injury to personnel. All

Support equipment:

<u>Nomenclature</u>	<u>PN</u>	<u>Specification</u>	<u>Qty</u>	<u>Task</u>
NA	--	--	--	--

Supplies:

<u>Nomenclature</u>	<u>PN</u>	<u>Specification</u>	<u>Qty</u>	<u>Task</u>
Sealant	--	MIL-PRF-81733, Type II-2	AR	4-2, 4-3

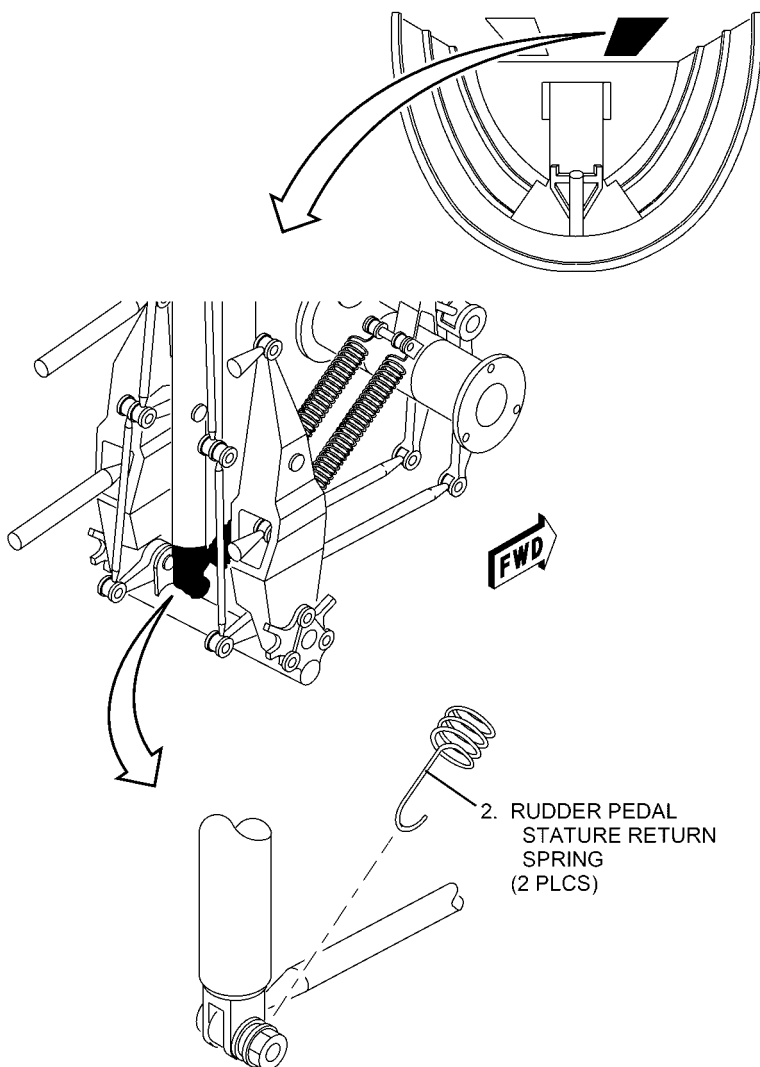
4-1. PREPARATION.

- 1. Review "Section 1 (General Information)" of this TO for system general warnings, cautions, and notes.
- 2. Review task "Functional Input Conditions" page for task specific safety conditions.
- 3. Install rudder aircraft ground safety lock (05-10-01, task 01-1).
- 4. Remove horizontal pressure panel access cover assembly (53-12-10).

CRANK REF DES	PANEL NO.	PANEL REF DES
2721AA015	112AZP	5312CA001
2721AA016	112BZP	5312CA002

**4-2. REPAIR RUDDER PEDAL STATURE ADJUST
CRANK ASSEMBLY BY REPLACING
UNIVERSAL JOINT.**

1. Perform interphone system operation (23-41-02, task 02-3).
2. (A,B) Disconnect rudder pedal stature return springs.



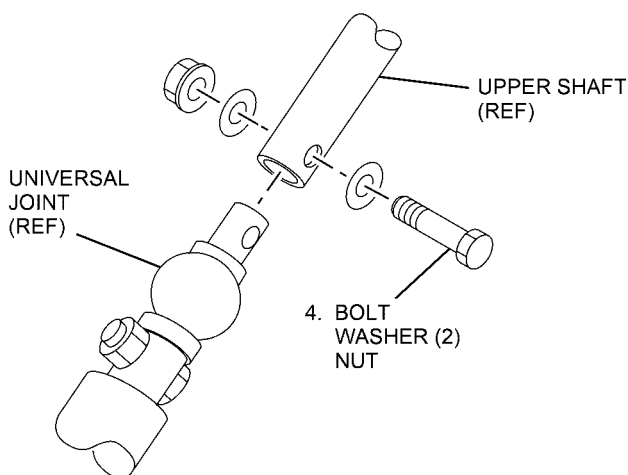
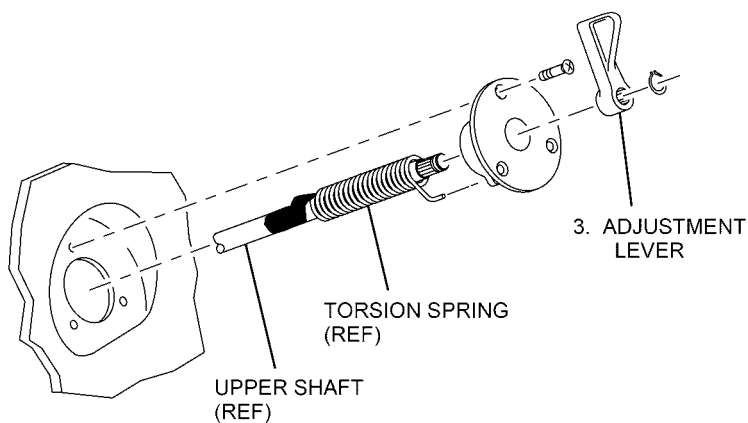
(TYPICAL)

ICN-88277-G2721501-002-01



Shaft torsion spring is hooked in back of rudder pedal adjustment support assembly and is preloaded. Support assembly shall be held in position when bolts are removed, then rotated clockwise to relieve tension on torsion spring. Failure to comply may cause damage to aircraft.

3. (A,B) Ensure rudder pedal adjustment lever is not preloaded by the upper shaft torsion spring.
4. (B) Remove nut, washers, and bolt from rudder pedal stature adjust upper shaft. Disconnect universal joint from upper shaft.

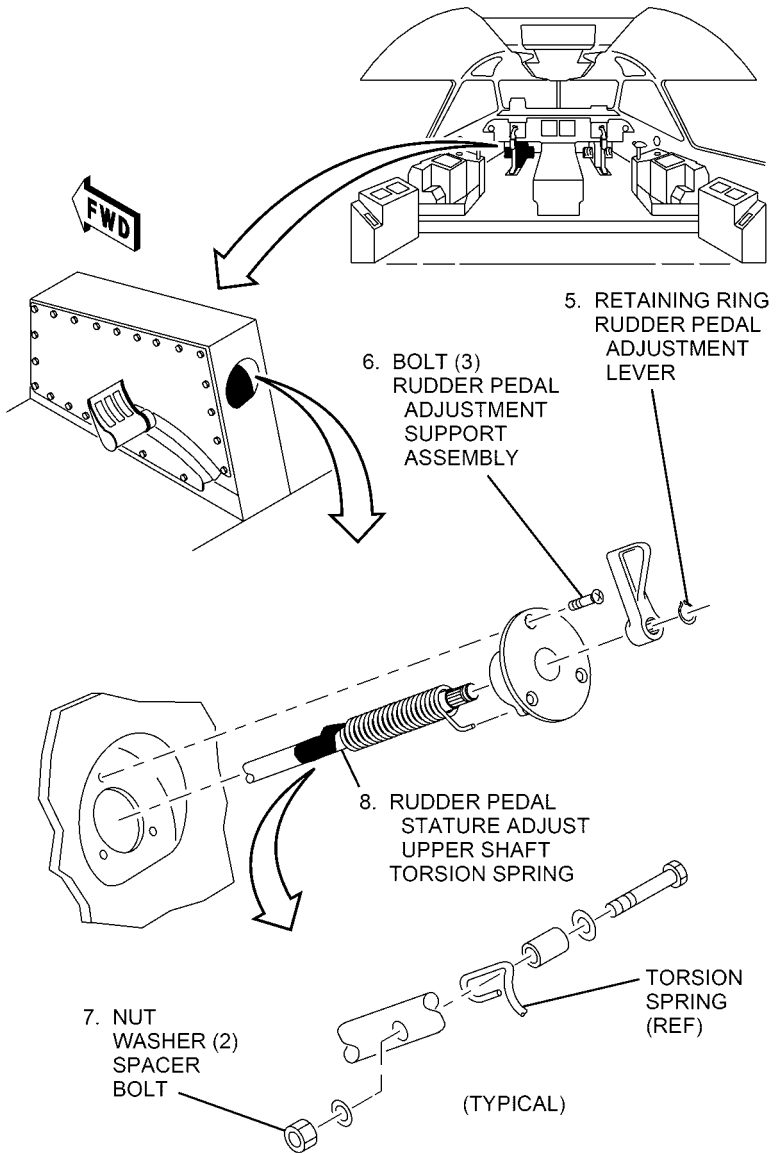


(TYPICAL)

ICN-88277-G2721502-002-01

TO 1300i-2-27JG-20-2

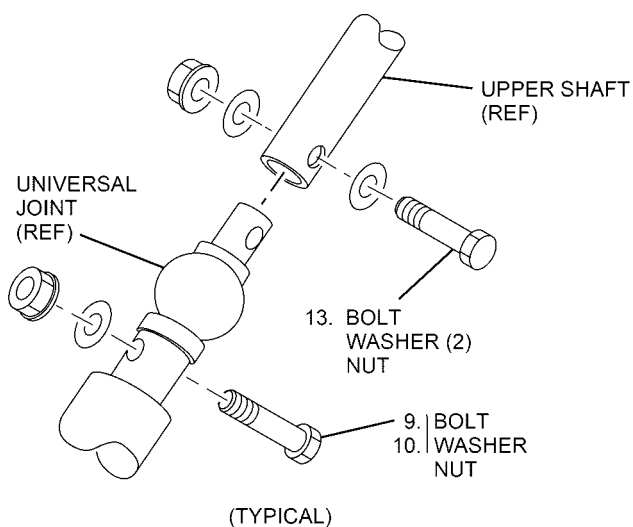
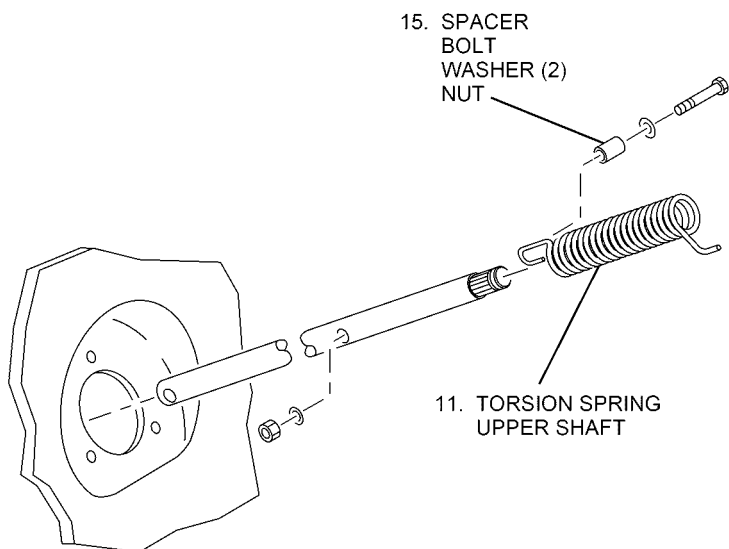
5. (A) Remove retaining ring and rudder pedal adjustment lever.
6. (A,B) Remove bolts; rotate rudder pedal adjustment support assembly one-half turn clockwise and remove support assembly.
7. (B) Remove nut, washers, spacer, and bolt attaching torsion spring.
8. (A,B) Remove rudder pedal stature adjust upper shaft and spring.



ICN-88277-G2721510-001-01

TO 1300i-2-27JG-20-2

9. (B) Identify and remove bolt, washer and nut. Disconnect universal joint from lower shaft.
10. (B) Connect the replacement universal joint to the lower shaft. Install bolt, washer and nut as identified.
11. (A,B) Slide torsion spring on upper shaft; position upper shaft over the universal joint on lower shaft.
12. Perform wet fastener installation (TO 1300i-23, Chapter 1, Section III).
13. (B) Install bolt, washers, and nut.
14. Perform wet fastener installation (TO 1300i-23, Chapter 1, Section III).
15. (B) Position spring; install spacer, bolt, washers, and nut.



ICN-88277-G2721511-001-01

TO 1300i-2-27JG-20-2

16. (A) Install support assembly on upper shaft.
17. (A,B) Position spring end in support assembly receptacle.

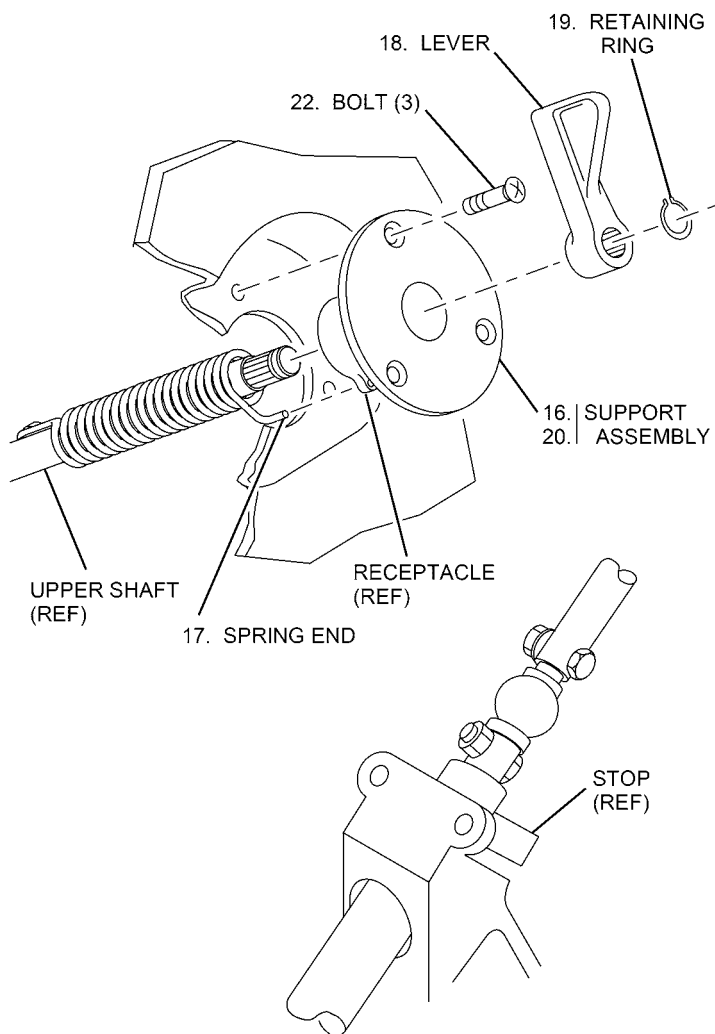
SPECIAL INSTRUCTION. Ensure spring end is inserted firmly in the support assembly receptacle.

18. (A,B) Position lever on shaft and rotate counter clockwise until stop engages.
19. (A) Verify lever is in vertical position and install retaining ring.

NOTE

Turn the support assembly until bolt holes are aligned.

20. (A) Hold lever in vertical position and rotate support assembly one-half turn counter clockwise.
21. Perform wet fastener installation (TO 1300i-23, Chapter 1, Section III).
22. (A) Install bolts.

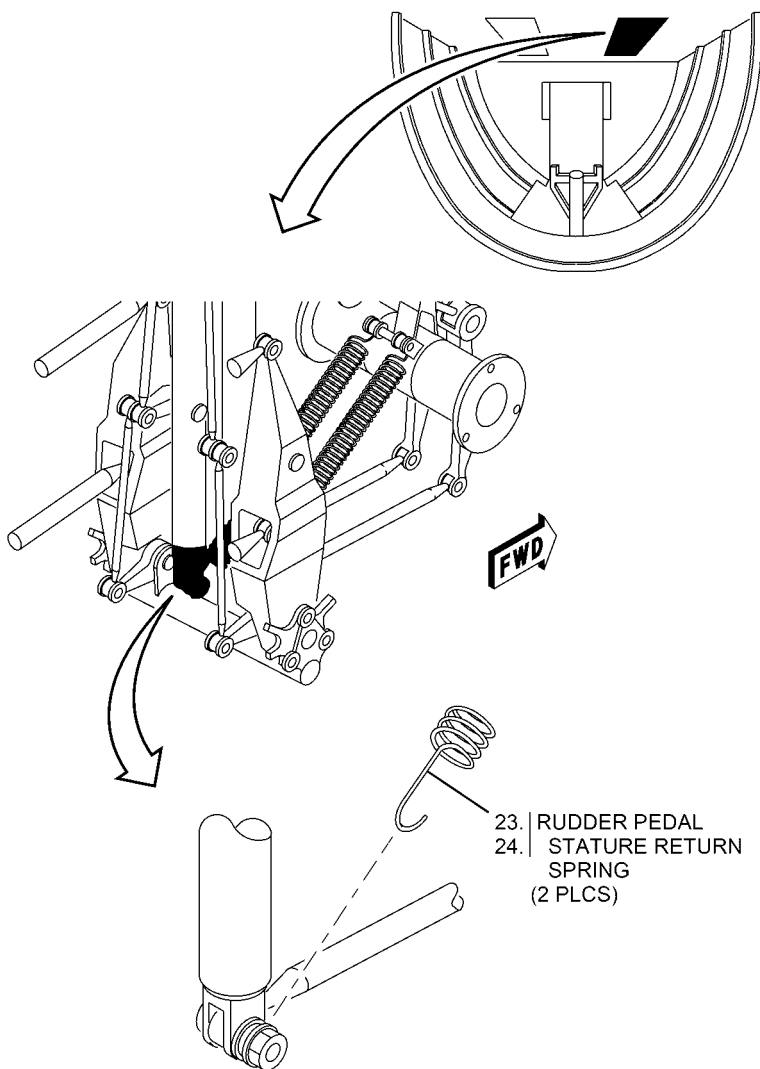


(TYPICAL)

ICN-88277-G2721512-001-01

TO 1300i-2-27JG-20-2

23. (A) Inspect rudder pedal stature return spring for cracks and general condition.
24. (A,B) Connect the rudder pedal stature return springs.
25. Perform interphone system operation (23-41-02, task 02-4).

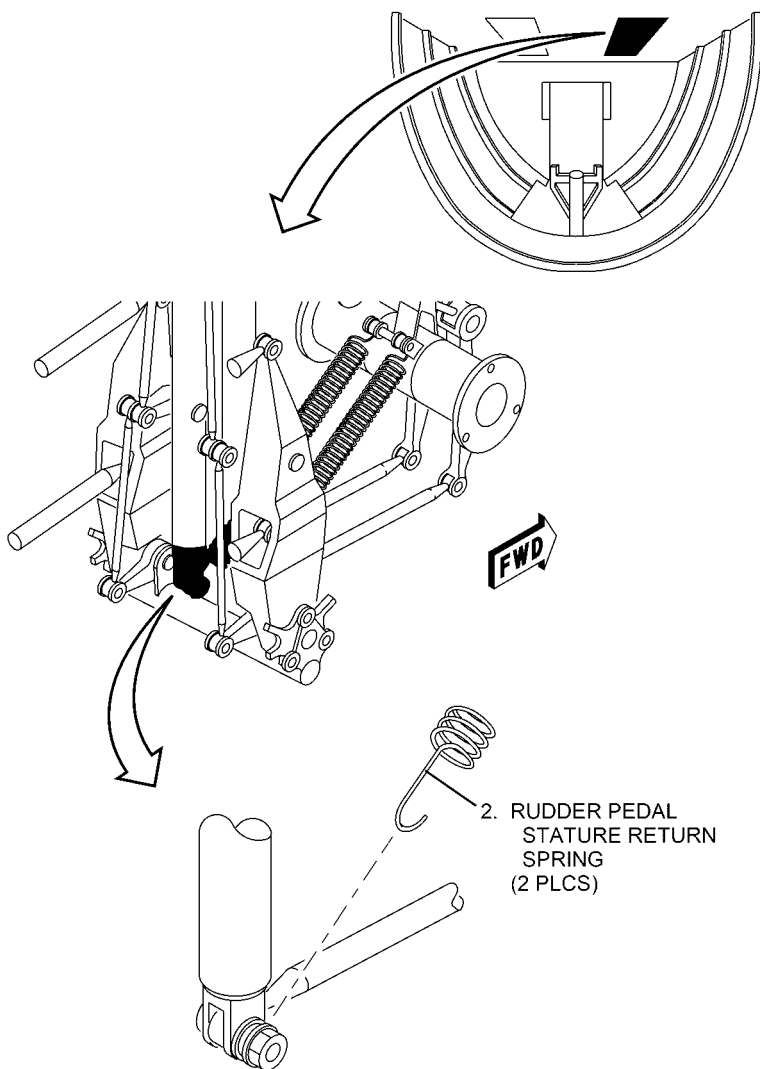


(TYPICAL)

ICN-88277-G2721503-002-01

**4-3. REPAIR RUDDER PEDAL STATURE ADJUST
CRANK ASSEMBLY BY REPLACING
ADJUSTMENT LEVER AND SHAFT.**

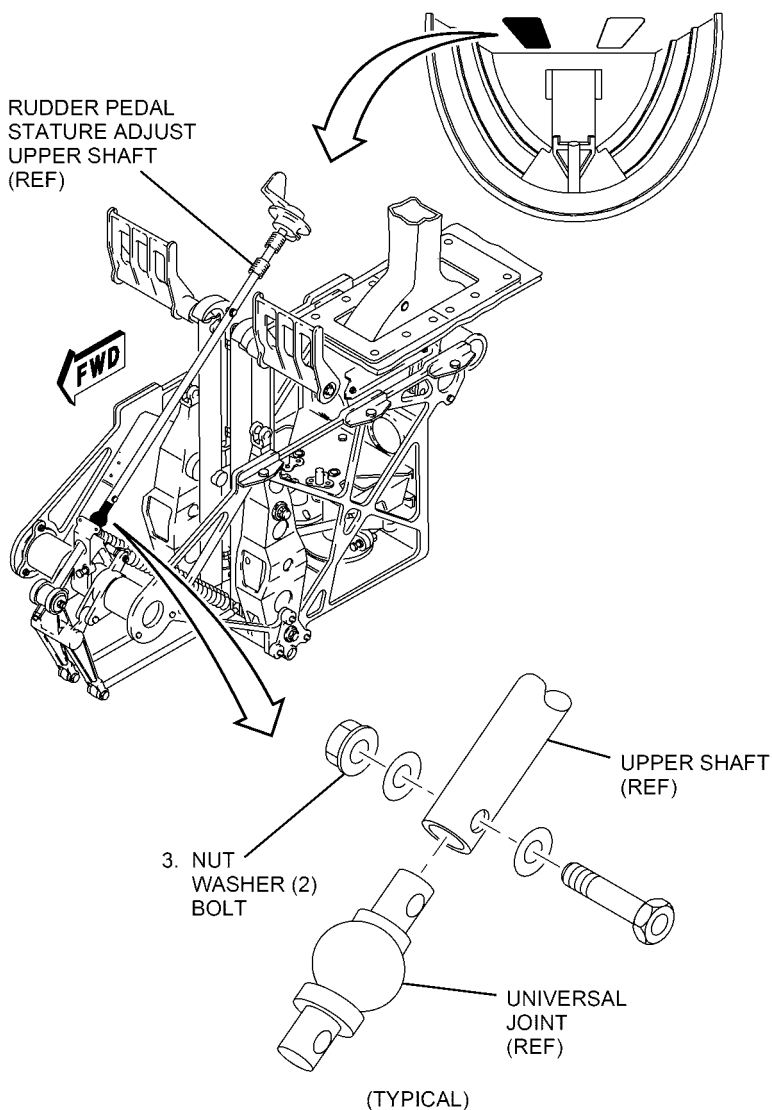
1. Perform interphone system operation (23-41-02, task 02-3).
2. (A) Disconnect rudder pedal stature return springs.



(TYPICAL)

ICN-88277-G2721504-001-01

3. (B) Remove nut, washers, and bolt from rudder pedal stature adjust upper shaft; disconnect universal joint from the upper shaft.



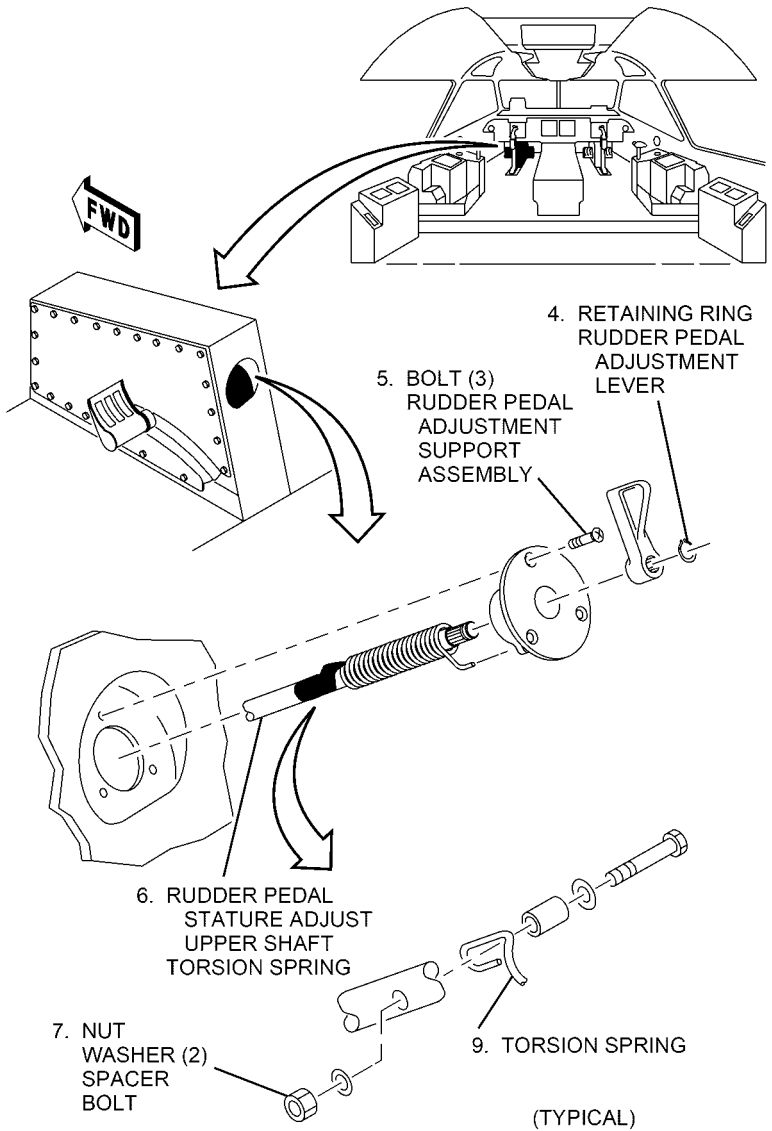
ICN-88277-G2721505-001-01

4. (A) Remove retaining ring and rudder pedal adjustment lever.



Shaft torsion spring is hooked in back of rudder pedal adjustment support assembly and is preloaded. Support assembly shall be held in position when bolts are removed, then rotated clockwise to relieve tension on torsion spring. Failure to comply may cause damage to aircraft.

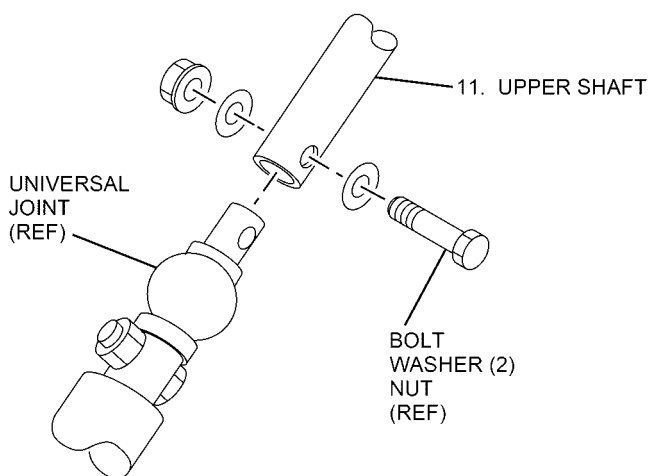
5. (A,B) Remove bolts; rotate rudder pedal adjustment support assembly one-half turn clockwise and remove support assembly.
6. (A,B) Remove rudder pedal stature adjust upper shaft and torsion spring.
7. (B) Remove nut, washers, spacer, bolt and disconnect torsion spring from the upper shaft.
8. Perform wet fastener installation (TO 1300i-23, Chapter 1, Section III).
9. (A) Connect torsion spring to the replacement shaft; install bolt, washers, spacer, and nut.



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TO 1300i-2-27JG-20-2

10. Perform wet fastener installation (TO 1300i-23, Chapter 1, Section III).
11. (A) Connect the replacement shaft to the universal joint; install bolt, washers, and nut.



(TYPICAL)

ICN-88277-G2721507-001-01

12. (A) Install support assembly on upper shaft.
13. (A) Position support assembly on spring end.

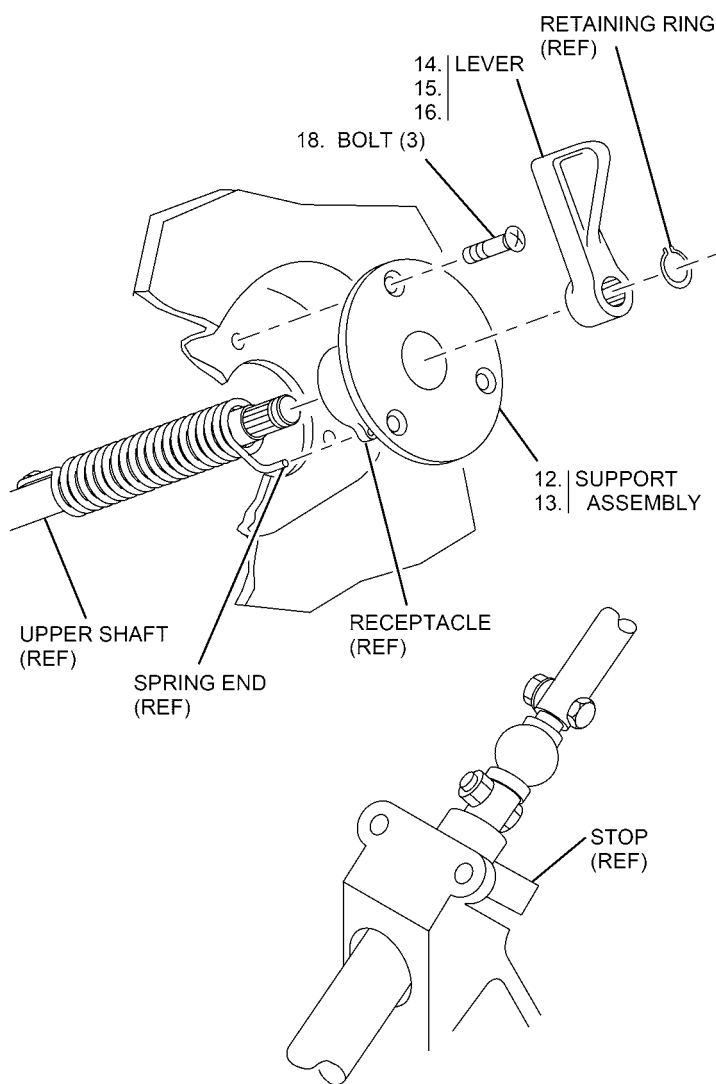
SPECIAL INSTRUCTION: Ensure spring end is inserted firmly in the support assembly receptacle.

14. (A,B) Position lever on shaft and rotate counter clockwise until stop engages.
15. (A) Verify lever is in vertical position and install retaining ring.

NOTE

Turn the support assembly until bolt holes are aligned.

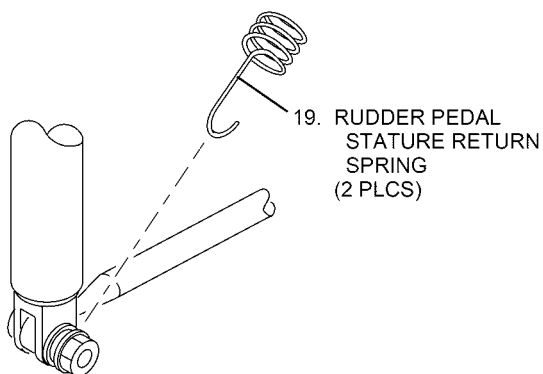
16. (A) Hold lever in vertical position and rotate support assembly one-half turn counter clockwise.
17. Perform wet fastener installation (TO 1300i-23, Chapter 1, Section III).
18. (A) Install bolts.



(TYPICAL)

ICN-88277-G2721514-001-01

19. (A) Connect the rudder pedal stature return springs.
20. Perform interphone system operation (23-41-02, task 02-3).



(TYPICAL)

ICN-88277-G2721508-001-01

4-4. FOLLOW-ON MAINTENANCE.

- 1. Remove rudder aircraft ground safety lock (05-10-01, task 01-2).
- 2. Perform rudder system operational checkout (27-20-01).
- 3. Install horizontal pressure panel access cover assembly (53-12-10).

CRANK REF DES	PANEL NO.	PANEL REF DES
2721AA015	112AZP	5312CA001
2721AA016	112BZP	5312CA002