TO 1300i-2-27JG-10-10

TECHNICAL MANUAL

JOB GUIDE
ORGANIZATIONAL MAINTENANCE

FLIGHT CONTROLS AILERON
(27-10-00 AND 27-13-11 THROUGH 27-15-10)

USAF SERIES
300i
AIRCRAFT

MCDONNELL DOUGLAS CORPORATION
MILITARY TRANSPORT AIRCRAFT
F33657-81-C-2108
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T-2 **1 SEPTEMBER 2024**

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TOTAL NUMBER OF PAGES IN THIS PUBLICATION IS 258 CONSISTING OF THE FOLLOWING:

Page No.	* Change No.	Page No.	* Change No.
Title thru T-2	0	2-101 thru 2-102	0
A	0	2-103 blank	0
B blank	0	2-104 thru 2-107	0
i	0	2-108 blank	0
ii blank	0	2-109 thru 2-125	0
iii thru iv	0	2-126 blank	0
1-1 thru 1-3	0	2-127 thru 2-128	0
1-4 blank	0	2-129 blank	0
2-1 thru 2-18	0	2-130 thru 2-172	0
2-19 blank	0	2-173 blank	0
2-20 thru 2-21	0	2-174 thru 2-193	0
2-22 blank	0	2-194 blank	0
2-23 thru 2-32	0	2-195 thru 2-196	0
2-33 blank	0	2-197 blank	0
2-34 thru 2-41	0	2-198 thru 2-245	0
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2-43 thru 2-46	0		
2-47 blank	0		
2-48 thru 2-81	0		
2-82 blank	0		
2-83	0		
2-84 blank	0		
2-85 thru 2-86	0		
2-87 blank	0		
2-88 thru 2-91	0		
2-92 blank	0		
2-93 thru 2-94	0		
2-95 blank	0		
2-96 thru 2-99	0		
2-100 blank	0		

*Zero in this column indicates an original page.

TABLE OF CONTENTS

INTRODUCTION

SCOPE.

This manual contains maintenance procedures for the operational checkout, removal, installation, maintenance built-in test, and adjustment of aileron system components.

MODEL(S) COVERED.

All

ABBREVIATIONS.

The following is a list of non-standard abbreviations used throughout this manual:

BLIN	BIT Logic Instruction
EFCS	Electronic Flight Control System
EPC	Electrical Power Center
IFCM	Integrated Flight Control Module
PLCS	Places
RVDT	Rotary Variable Differential Transducer

CHANGE REQUEST.

Recommended changes to this manual shall be submitted in accordance with TO 00-5-1.

300i TO INFORMATION.

General 300i TO/eTO, TO Manager, Supplement and finalized Recommended Change (RC) information can be found in the Enhanced Technical Information Management System (ETIMS), System of Record.

LIST OF TIME COMPLIANCE TECHNICAL ORDERS (TCTO).

This list of TCTO's contains all current TCTO's that affect the technical content of text or illustrations found in this manual.

TCTO NUMBER	TITLE	TCTO DATE	APPLICABILITY
1300i-2393	Replacement of Head Up Display from Part No. 17B1U6015-525 or 17B1U6015-527 to Part No. 4442-1000-10 (Ref Des 3425AA001 and 3425AA002), 300i Aircraft	17 MAR 21	AA → MM

SECTION 1

GENERAL INFORMATION (27-10-00)

1-1. GENERAL INFORMATION.

1-2. This section provides general information that is essential for ensuring complete and safe maintenance procedures contained throughout this job guide manual.

1-3. When operating an auxiliary motor pump below 15 degrees Fahrenheit a 30 seconds on/30 seconds off duty cycle for a maximum 10 cycles may be required to reach full hydraulic pressure of 3800 to 4200 psi. Allow ten minutes for cooling and repeat cycles.

1-4. Hydraulic system No. 2 may require 45 seconds before reaching full hydraulic pressure of 3800 to 4200 psi.

1-5. When reading the difference between pressurized and unpressurized reservoir quantities, the quantities shall be no greater than the maximum allowed and as follows:

HYD SYSTEM NO.	QTY DIFFERENCE (Quarts)
1	2
2	4
3	3
4	2

1-6. Flight control surfaces are to be cleared prior to turning off hydraulic auxiliary pumps from the loadmaster control panels. Flight control surface movement may occur.

1-7. All adhesive sealants, sealants, and compounds used in this manual are listed with a primary part number and/or primary specification number. Any suitable substitutes and/or interchangeable adhesive sealants, sealants, and compounds may be used unless otherwise specified. Suitable substitutes and/or interchangeable adhesive sealants, sealants, and compounds are listed in the system peculiar corrosion control manual (Refer to TO 1300i-23, Chapter 1, Section III).

1-7A. Proper tube clearances shall be maintained after hydraulic component replacement, where loosening of tube support clamps and/or brackets on interfacing hydraulic lines is necessary to relieve

tube-to-fitting insertion and facilitate component removal. Minimum allowable hydraulic tubing clearances may be found in the system's General System (GS) manual (Refer to TO 1300i-2-29GS-00-1, Chapter 2).

1-8. GENERAL WARNINGS, CAUTIONS, AND NOTES.

WARNING

- All flight control surfaces and thrust reversers shall be clear of personnel and equipment prior to applying or removing hydraulic power. Failure to comply may cause injury to personnel or damage to aircraft.

- All flight control surfaces and thrust reversers shall be clear of personnel and equipment prior to any movement of the surfaces. Failure to comply may cause injury to personnel or damage to aircraft.

WARNING - Continued

- Warning tags shall be attached to all opened circuit breakers as directed by the technical order and an applicable warning statement shall be entered in the AFTO Form/IMT 781A, IAW TO 00-20-1, AFI 21-101 and TO 1300i-2-00GV-00-1, Chapter 5 anytime exiting cockpit circuit breaker area before task completion or there is a delay in maintenance where task cannot be fully accomplished. Failure to comply may cause injury to personnel or damage to aircraft.

CAUTION

Air in a hydraulic system can cause numerous malfunctions, from a total system failure to a minor indication problem. When you suspect air has been inducted into a system by removing a hydraulic component or a line, refer to the hydraulic system bleed procedure, (12-29-08). Failure to comply may cause damage to aircraft.

NOTE

When installing or removing rig pins during the rigging of cables, the pins shall move freely in rig pin holes.

SECTION 2

AILERON INTEGRATED FLIGHT CONTROL MODULE (27-13-11)

MASTER INPUT CONDITIONS:

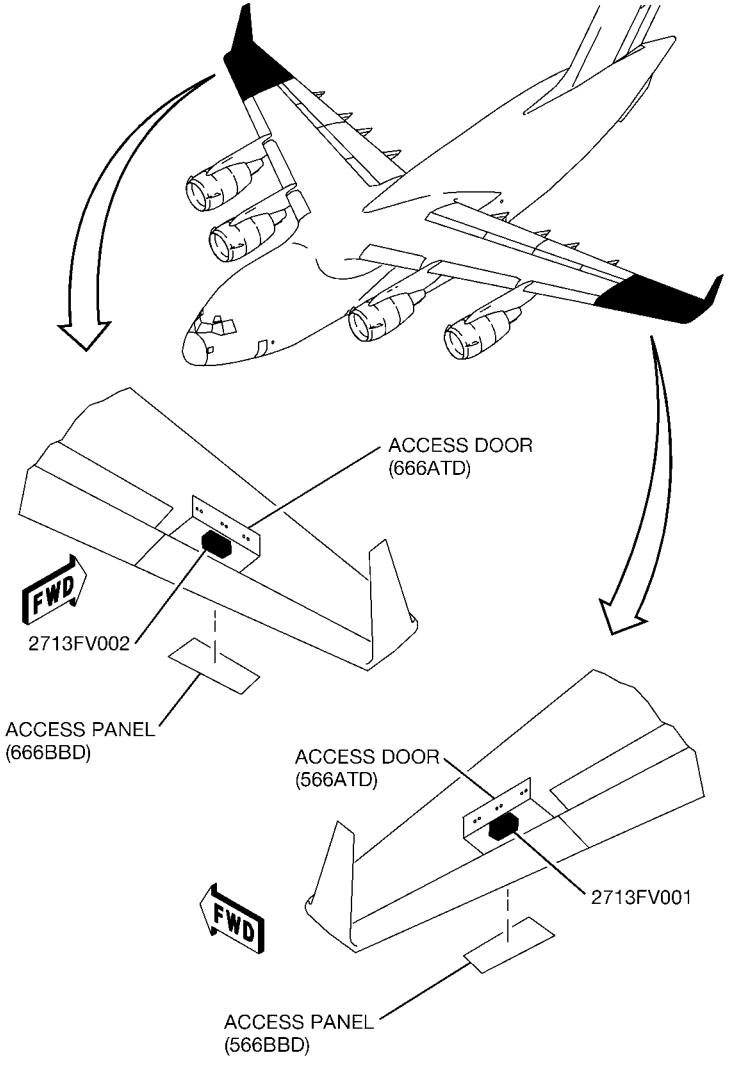
Reference designators:

2713FV001	Left Aileron Integrated Flight Control Module
2713FV002	Right Aileron Integrated Flight Control Module

Applicable functions:

-2 Removal.
-3 Installation.
-4 Repair.

Access data:



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AILERON INTEGRATED FLIGHT CONTROL MODULE REMOVAL (27-13-11-2)

FUNCTIONAL INPUT CONDITIONS:

Applicability:

	Task
All	All

Additional information:

This procedure consists of the following tasks:

2-1. Preparation.
2-2. Removal.

NOTE

	Task
• Hydraulic system components and lines contain unknown quantities of fluid.	2-2
• This is a typical removal task for all aileron integrated flight control modules.	All

Additional data:

	Task
TO 1300i-2-00JG-00-1	2-1
TO 1300i-2-05JG-10-1	2-1
TO 1300i-2-10JG-10-1	2-1
TO 1300i-2-12JG-29-1	2-1
TO 1300i-2-57JG-50-1	2-1

Personnel recommended:

	Task
One	2-1
Two	2-2

	Task
Person (A) performs task.	
Person (B) assists person (A).	

Safety conditions:

	Task
WARNING	
Areas below lower wing fixed trailing edge door assembly shall remain clear of personnel when door is in lowered position and suspended from lanyards. Failure to comply may cause injury to personnel or damage to aircraft.	All

Support equipment:

Nomenclature	PN	Specification	Qty	Task
Knife, Putty	PK30	--	AR	2-2
Pail, Utility	B12R	--	AR	2-2

Supplies:

Nomenclature	PN	Specification	Qty	Task
Tag, Warning	--	--	4	2-1

2-1. PREPARATION.

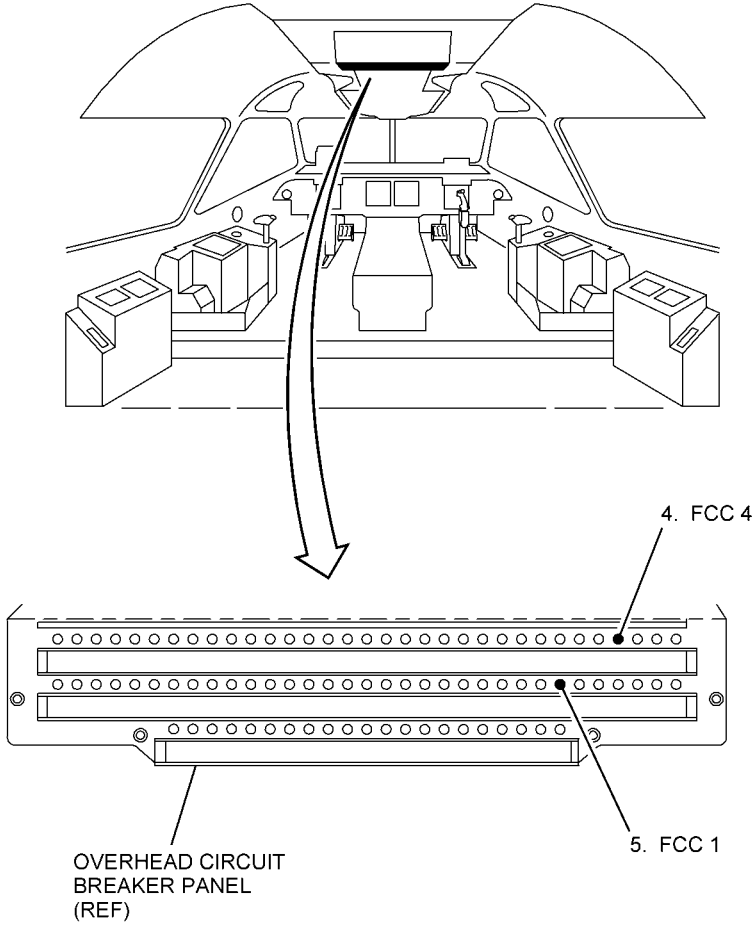
1. Review ″Section 1 (General Information)″ of this TO for system general warnings, cautions, and notes.

2. Review task ″Functional Input Conditions″ page for task specific safety conditions.

3. Install aileron control surface aircraft ground safety lock (05-10-01, task 01-9).

4. Open **FCC 4** circuit breaker on overhead circuit breaker panel, row **G**, column **30**, and attach warning tag.

5. Open **FCC 1** circuit breaker on overhead circuit breaker panel, row **H**, column **27**, and attach warning tag.



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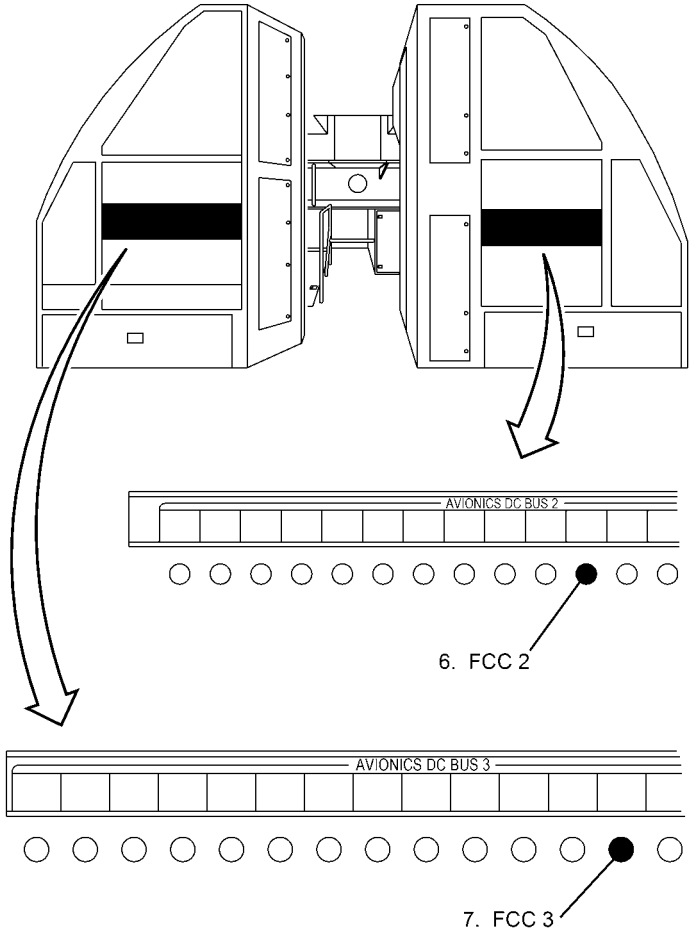
6. Open **FCC 2** circuit breaker on Electrical Power Center (EPC), row **T**, column **50** and attach warning tag.

7. Open **FCC 3** circuit breaker on EPC, row **T**, column **24** and attach warning tag.



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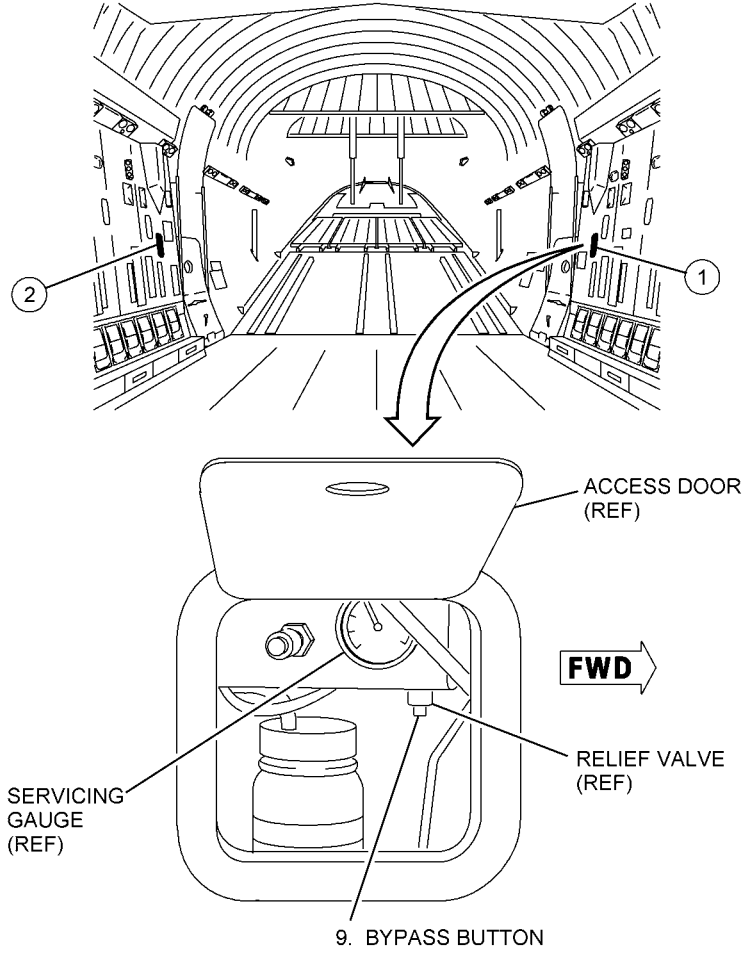
CAUTION

When winds are above 14 knots, aircraft ground safety locks shall be installed per (10-10-00, para 1-12) prior to bleeding down the gust damper system(s). Failure to comply may cause damage to control surfaces.

8. Open access door.

9. Press bypass button on hydraulic system relief valve until servicing gauge stops decreasing.

   - Servicing gauge reads nitrogen precharge (TO 1300i-2-12JG-29-3, 12-29-00, para 1-21).



8.

MODULE REF DES	INDEX NO.	HYDRAULIC SYSTEM	ACCESS DOOR NO.
2713FV001	1	2	265TZM (AT)
2713FV002	2	3	266TZM (AT)

ICN-88277-G2713037-005-01

10. Deleted.

NOTE

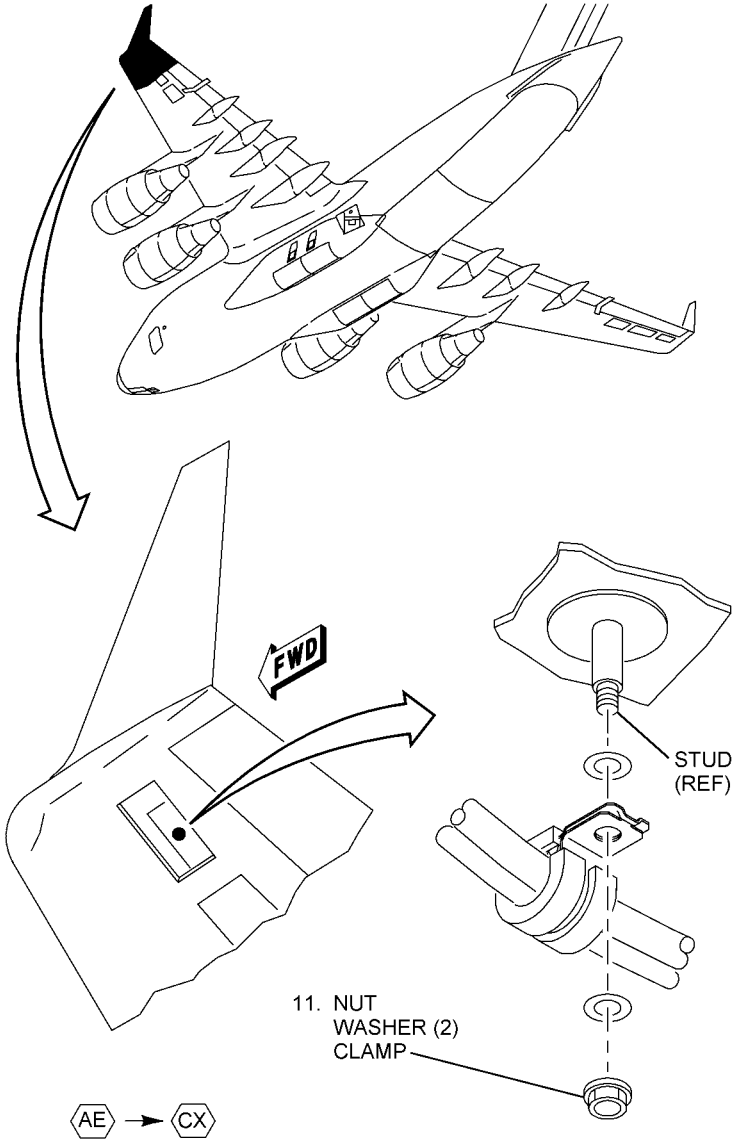
Lower wing fixed trailing edge door assembly (57-54-01) removed during installation of the aileron control surface aircraft ground safety lock, (05-10-01, task 01-9).

IFCM REF DES	DOOR NO.	DOOR REF DES
2713FV001	566BBD	5754CA025
2713FV002	666BBD	5754CA040

11. ⟨AE⟩ → ⟨CX⟩ Remove nut, washers, and clamp from stud.

11. ⟨AA⟩ → ⟨AD⟩ No action required.

11. ⟨CY⟩ → No action required.



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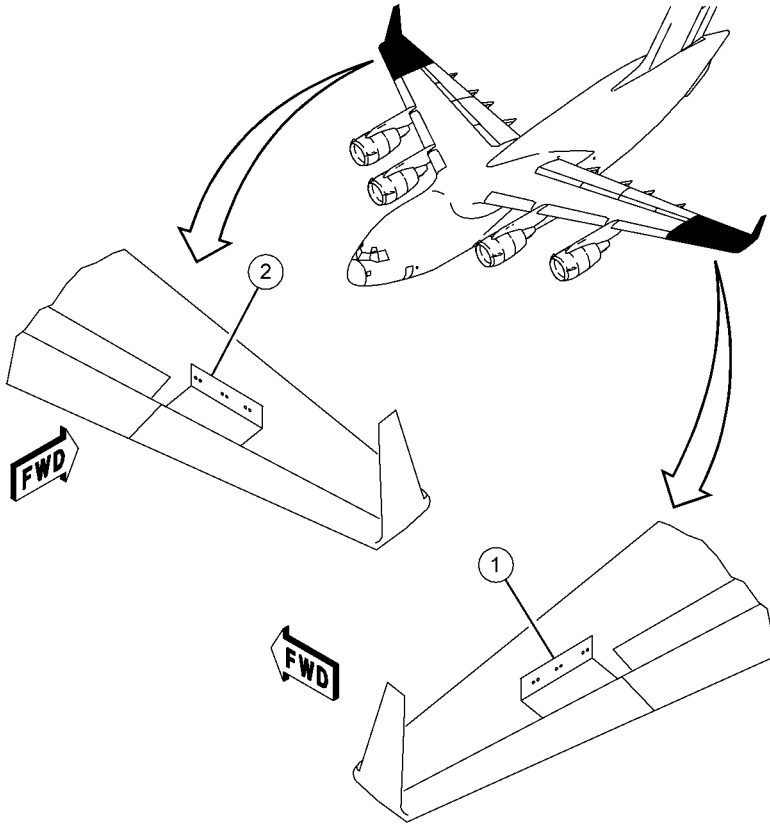
12. Enter fuselage and wing (00-00-01).

13. Loosen fasteners and open access door.



13.

IFCM REF DES	INDEX NO.	DOOR NO.	FASTENER QTY
2713FV001	1	566ATD	6
2713FV002	2	666ATD	6

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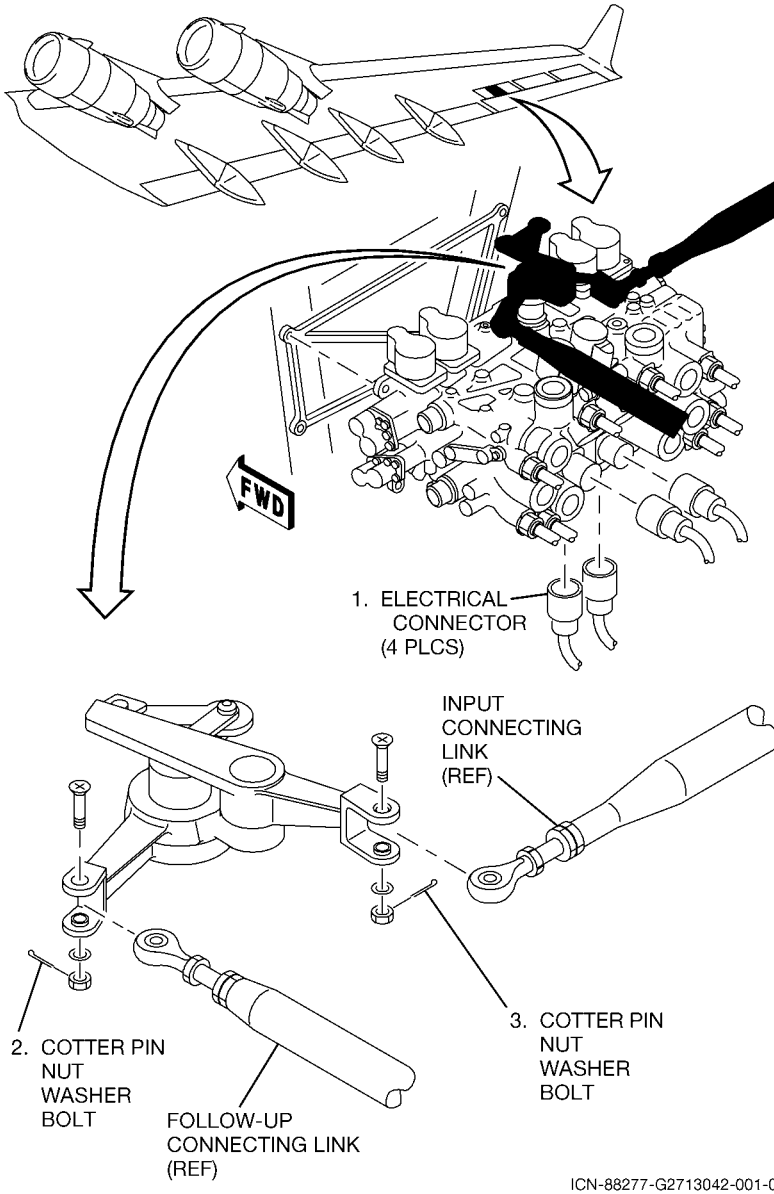
2-2. REMOVAL.

1. (A) Disconnect electrical connectors.

2. (A) Remove cotter pin, nut, washer, and bolt from follow-up connecting link.

3. (A) Remove cotter pin, nut, washer, and bolt from input connecting link.



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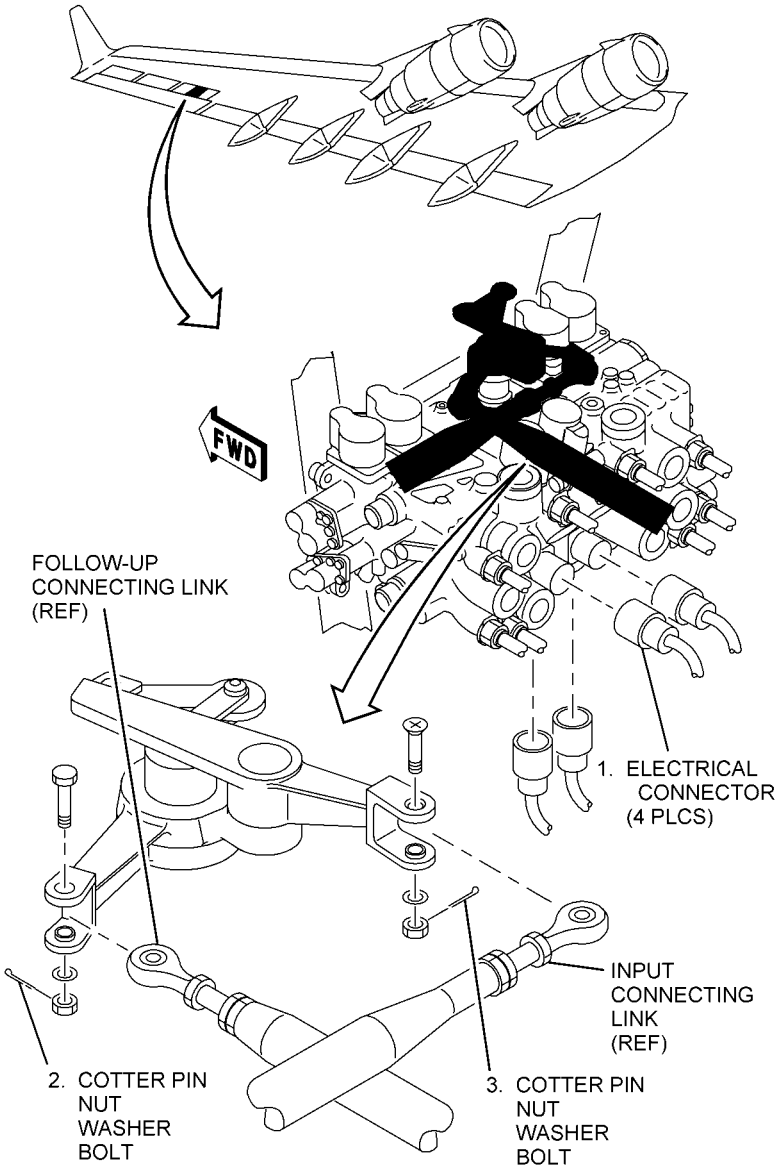

FWD
FOLLOW-UP
CONNECTING LINK
(REF)
1. ELECTRICAL
CONNECTOR
(4 PLCS)
INPUT
CONNECTING
LINK
(REF)
2. COTTER PIN
NUT
WASHER
BOLT
3. COTTER PIN
NUT
WASHER
BOLT
ICN-88277-G2713026-005-01

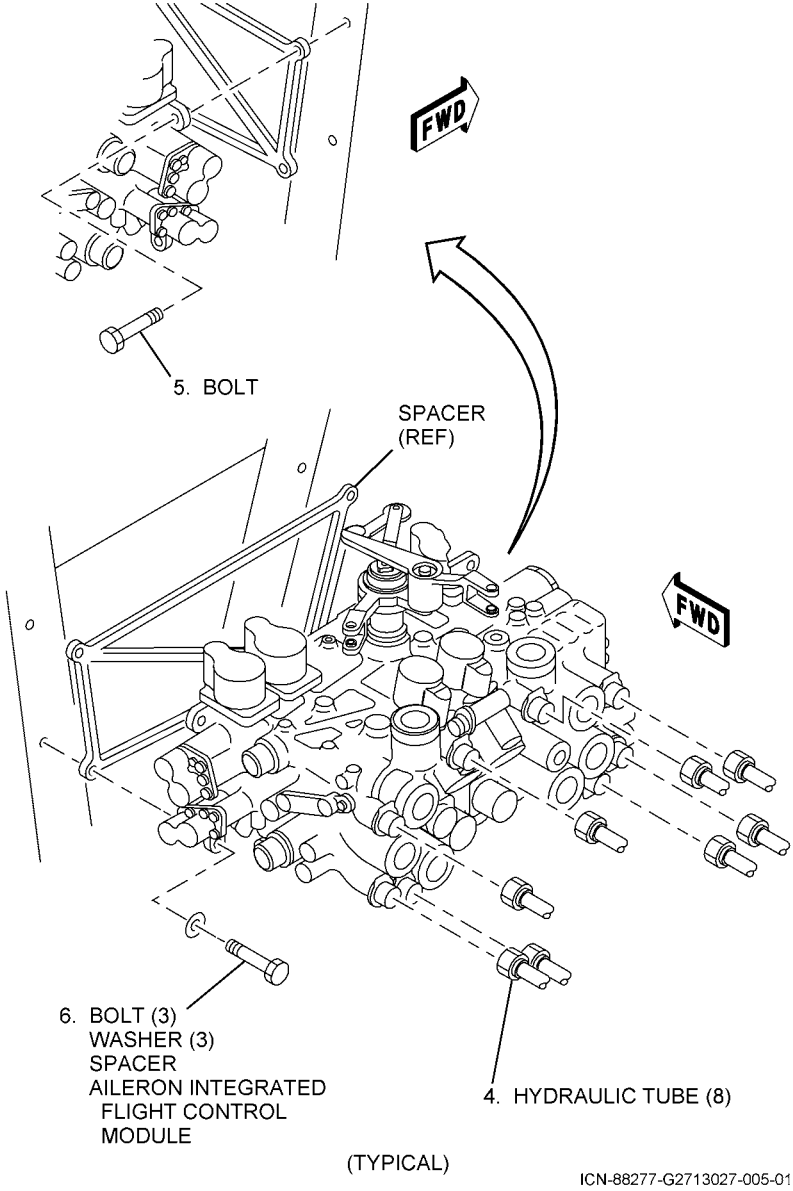
4. (A) Disconnect hydraulic tubes.

5. (A,B) Remove bolt.

CAUTION

Integrated Flight Control Module (IFCM) may be bonded to supporting structure with sealant. Carefully separate IFCM mounting pads from structure using a thin metal putty knife. Do not pry or bend mounting pads. Failure to comply may cause damage to aircraft or equipment.

6. (A,B) Remove bolts, washers, spacer, and aileron (IFCM).



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AILERON INTEGRATED FLIGHT CONTROL MODULE INSTALLATION (27-13-11-3)

FUNCTIONAL INPUT CONDITIONS:

Applicability:

	Task
All	All

Additional information:

This procedure consists of the following tasks:

3-1. Installation.
3-2. Follow-on maintenance.

NOTE	Task
This is a typical installation task for all aileron integrated flight control modules.	All

Additional data:

	Task
TO 1300i-2-00JG-00-1	3-2
TO 1300i-2-05JG-10-1	3-2
TO 1300i-2-22JG-10-1	3-2
TO 1300i-2-27JG-10-1	3-2
TO 1300i-2-40JG-00-1	3-2
TO 1300i-2-57JG-50-1	3-2
TO 1300i-23	3-1

Personnel recommended:

	Task
One	3-2
Two	3-1

	Task
Person (A) performs task.	
Person (B) assists person (A).	

Safety conditions:

	Task
WARNING	
Areas below lower wing fixed trailing edge door assembly shall remain clear of personnel when door is in lowered position and suspended from lanyards. Failure to comply may cause injury to personnel or damage to aircraft.	All

Support equipment:

Nomenclature	PN	Specification	Qty	Task
Wrench, Torque	--	(0-150 in-lb)	1	3-1
Wrench, Torque	--	(150-750 in-lb)	1	3-1

Supplies:

Nomenclature	PN	Specification	Qty	Task
Pin, Cotter	MS24665-151	--	2	3-1
Sealant	PR-1775 B-1/2	AMS 3265	AR	3-1

3-1. INSTALLATION.

1. Review ″Section 1 (General Information)″ of this TO for system general warnings, cautions, and notes.

2. Review task ″Functional Input Conditions″ page for task specific safety conditions.

3. Perform class R electrical bonding on lower right mounting pad and mating surface (TO 1300i-23, Chapter 1, Section III).

4. Perform wet fastener installation (TO 1300i-23, Chapter 1, Section III).

5. (A,B) Position spacer and aileron Integrated Flight Control Module (IFCM).

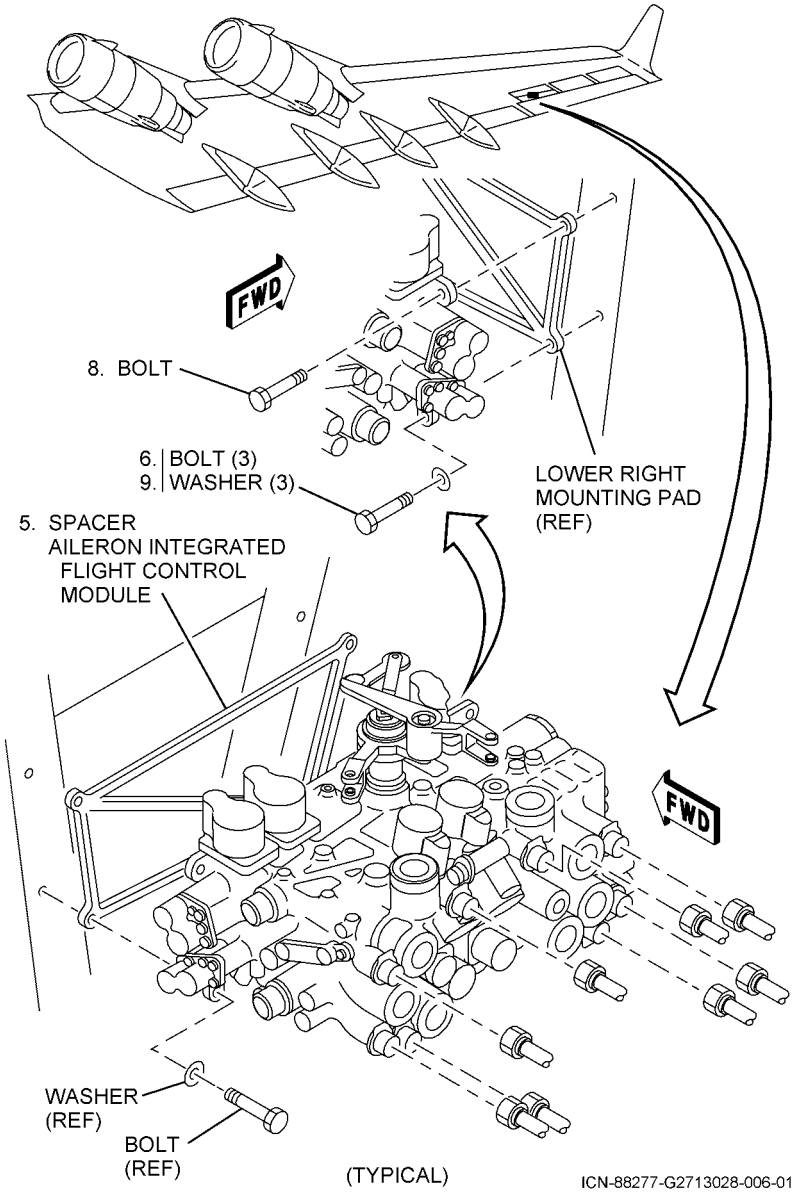
6. (A) Install washers and bolts; torque **100-120 in-lb**.

7. Perform wet fastener installation (TO 1300i-23, Chapter 1, Section III).

8. (A) Install bolt and torque **100-120 in-lb**.

9. (A) Retorque lower right bolt **100-120 in-lb**.

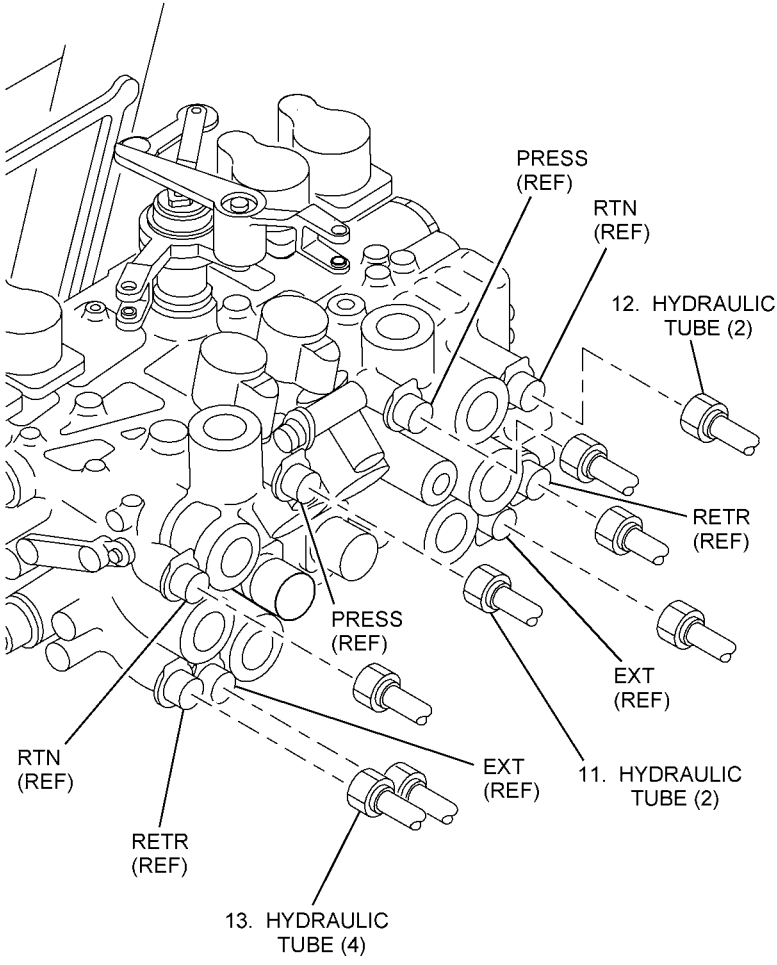
10. (B) Measure the class R electrical bonding resistance values from component to structure (TO 1300i-23, Chapter 1, Section III).


FWD
8. BOLT
6. BOLT (3)
9. WASHER (3)
LOWER RIGHT
MOUNTING PAD
(REF)
5. SPACER
AILERON INTEGRATED
FLIGHT CONTROL
MODULE
FWD
WASHER
(REF)
BOLT
(REF)
(TYPICAL)
ICN-88277-G2713028-006-01

11. (A) Connect hydraulic tubes to **PRESS** port and torque **470-550 in-lb**.

12. (A) Connect hydraulic tubes to **RTN** port and torque **620-700 in-lb**.

13. (A) Connect hydraulic tubes to **EXT** and **RETR**; torque **215-245 in-lb**.



(TYPICAL)

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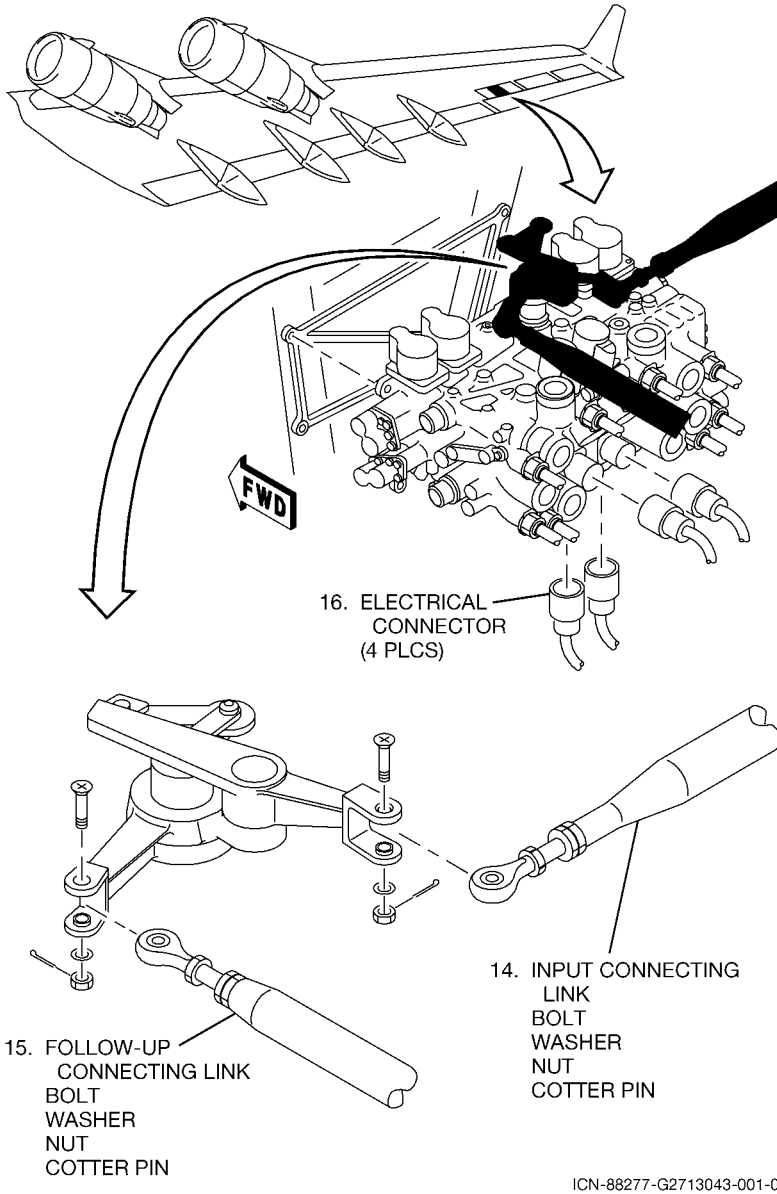
14. (A) Position input connecting link; install bolt, washer, nut, and cotter pin.

15. (A) Position follow-up connecting link; install bolt, washer, nut, and cotter pin.

16. (A) Connect electrical connectors as follows:

IFCM REF DES	CONNECTOR REF DES	JUNCTION
2713FV001	2713PP001	J1
	2713PP002	J2
	2713PP003	J3
	2713PP004	J4
2713FV002	2713PP005	J1
	2713PP006	J2
	2713PP007	J3
	2713PP008	J4



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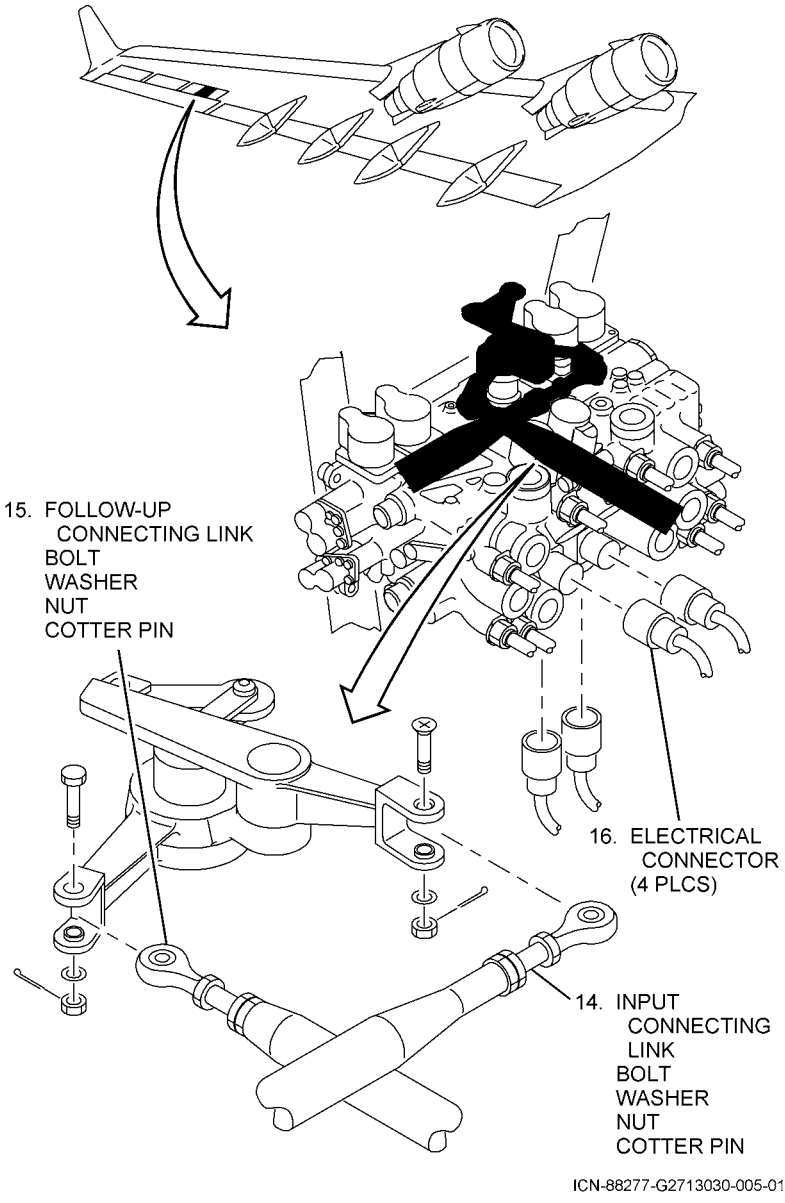
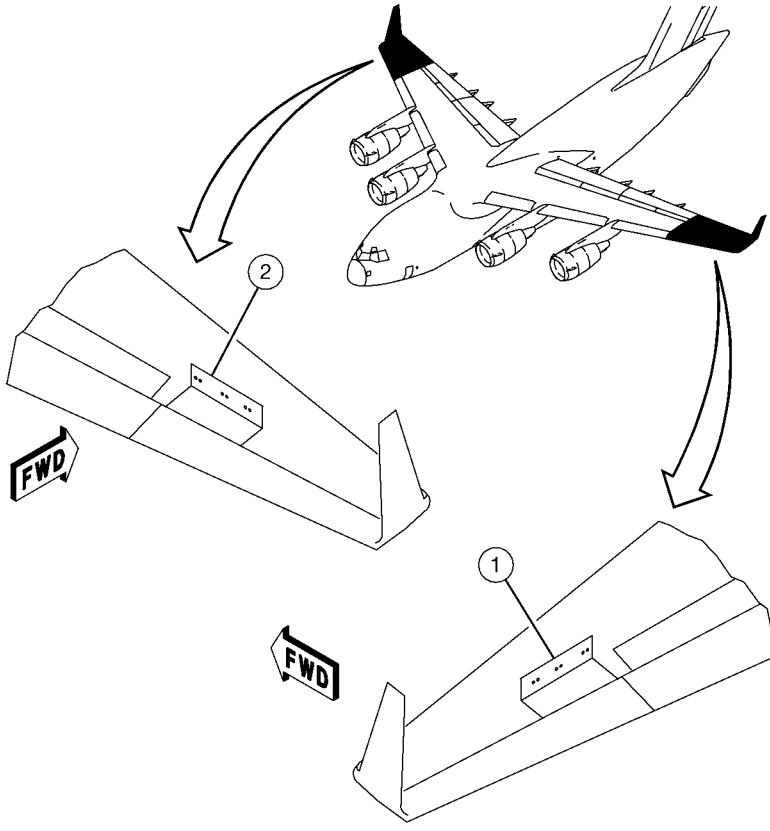

15. FOLLOW-UP
CONNECTING LINK
BOLT
WASHER
NUT
COTTER PIN
16. ELECTRICAL
CONNECTOR
(4 PLCS)
14. INPUT
CONNECTING
LINK
BOLT
WASHER
NUT
COTTER PIN
ICN-88277-G2713030-005-01

3-2. FOLLOW-ON MAINTENANCE.

1. Close access door, and tighten fasteners.

2. Exit fuselage and wing (00-00-01).



1.

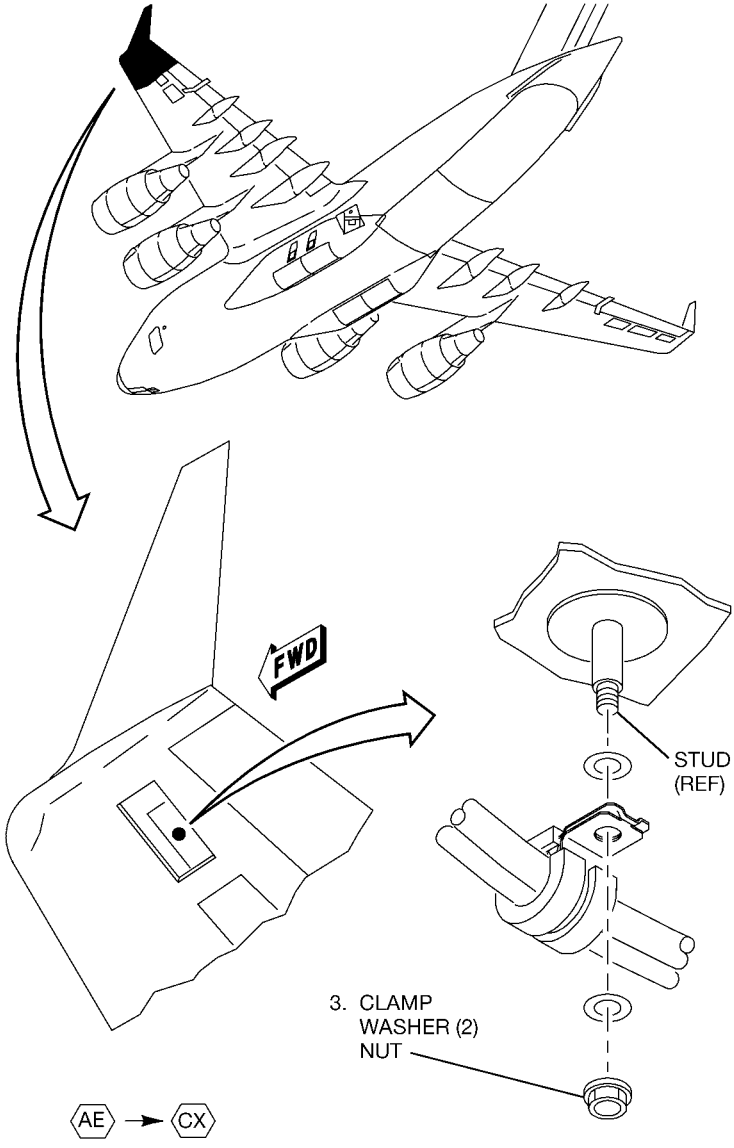
IFCM REF DES	INDEX NO.	DOOR NO.	FASTENER QTY
2713FV001	1	566ATD	6
2713FV002	2	666ATD	6

ICN-88277-G2713036-005-01

3. ⟨AA⟩ → ⟨AD⟩ No action required.

3. ⟨AE⟩ → ⟨CX⟩ Install washer on stud; position clamp, and install washer and nut; tighten nut.

3. ⟨CY⟩ → No action required.



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4. Deleted.

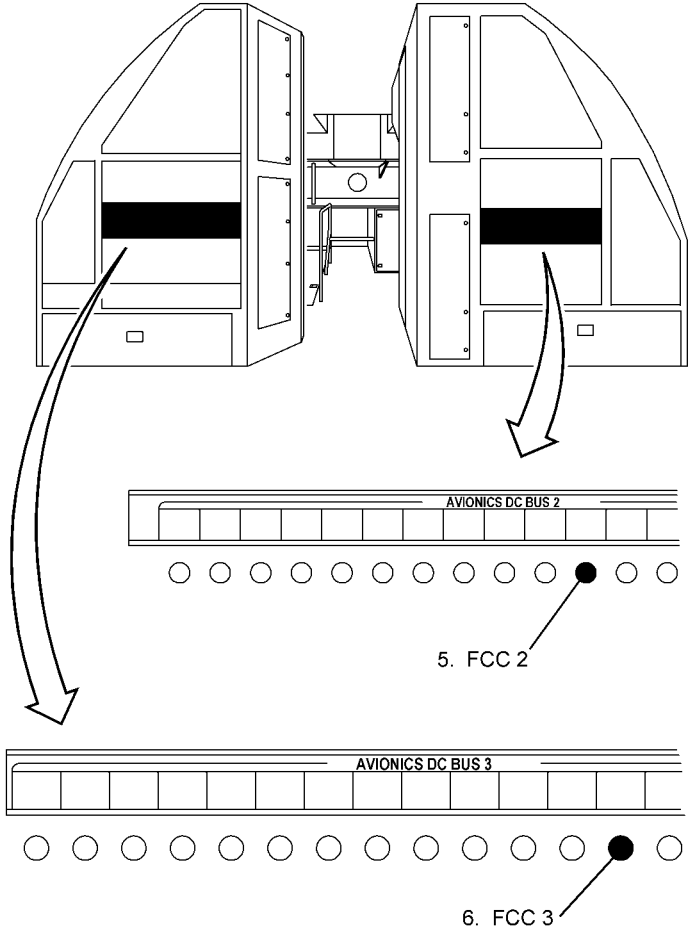
NOTE

Lower wing fixed trailing edge door assembly (57-54-01) installed during removal of the aileron control surface aircraft ground safety lock, (05-10-01, task 01-10).

IFCM REF DES	DOOR NO.	DOOR REF DES
2713FV001	566BBD	5754CA025
2713FV002	666BBD	5754CA040

5. Remove warning tag and close **FCC 2** circuit breaker on Electrical Power Center (EPC), row **T**, column **50**.

6. Remove warning tag and close **FCC 3** circuit breaker on EPC, row **T**, column **24**.



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7. Remove warning tag and close **FCC 4** circuit breaker on overhead circuit breaker panel, row **G**, column **30**.

8. Remove warning tag and close **FCC 1** circuit breaker on overhead circuit breaker panel, row **H**, column **27**.

9. Remove aileron control surface aircraft ground safety lock (05-10-01, task 01-10).

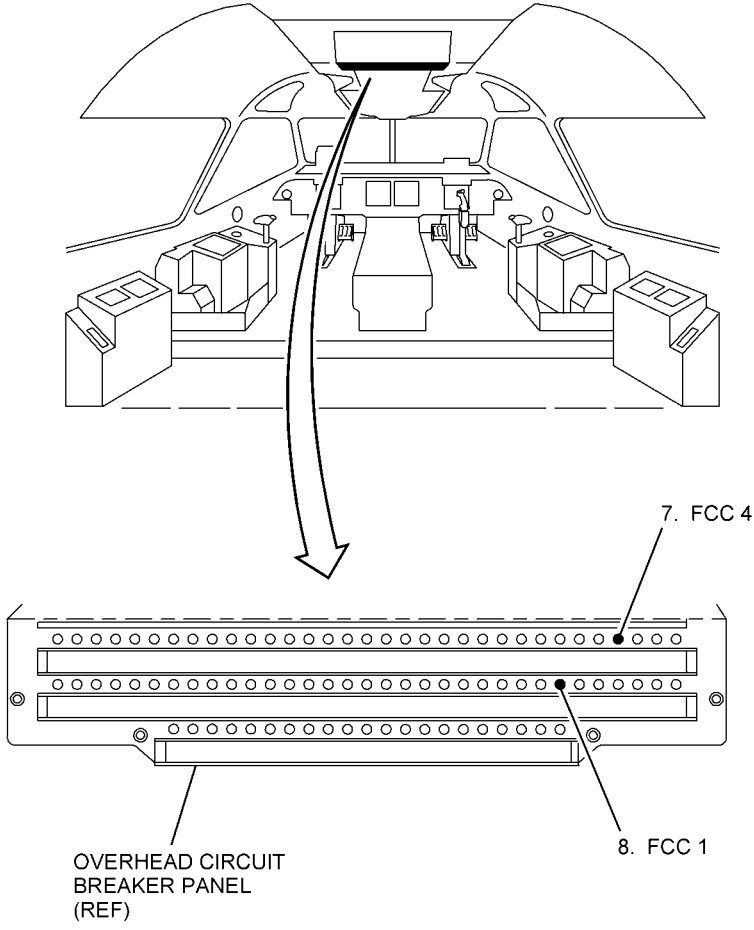
CAUTION

Steps 10, 11, and 12 must be done in order with each step completed before going on to the next step. Failure to comply may cause damage to aircraft.

10. Perform aileron mechanical controls and surfaces system operational checkout (27-11-01, task 01-1).

11. Perform electronic flight control system maintenance built-in test (40-00-02).

12. Perform electronic flight control system minor rig (22-10-02).



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AILERON INTEGRATED FLIGHT CONTROL MODULE REPAIR (27-13-11-4)

FUNCTIONAL INPUT CONDITIONS:

Applicability:

	Task
All	All

Additional information:

This procedure consists of the following task:

4-1. Repair aileron integrated flight control module by replacing quad contact switch.

	Task
NOTE This is a typical repair task for all aileron integrated flight control modules quad contact switches.	All

Additional data:

		Task
TO	1300i-2-00JG-00-1	All
TO	1300i-2-05JG-10-1	All
TO	1300i-2-40JG-00-1	All
TO	1300i-2-57JG-50-1	All

Personnel recommended:

	Task
One	All

Safety conditions:

	Task
WARNING	
Areas below lower wing fixed trailing edge door assembly shall remain clear of personnel when door is in lowered position and suspended from lanyards. Failure to comply may cause injury to personnel or damage to aircraft.	All

Support equipment:

Nomenclature	PN	Specification	Qty	Task
Wrench, Torque	--	(0-25 in-lb)	1	All

Supplies:

Nomenclature	PN	Specification	Qty	Task
Packing, Preformed	M83461/1-014	--	AR	All
Packing, Preformed	M83461/1-015	--	AR	All
Packing, Preformed	M83461/1-016	--	AR	All
Tag, Warning	--	--	6	All
Wire, Safety	MS20995C20	--	AR	All

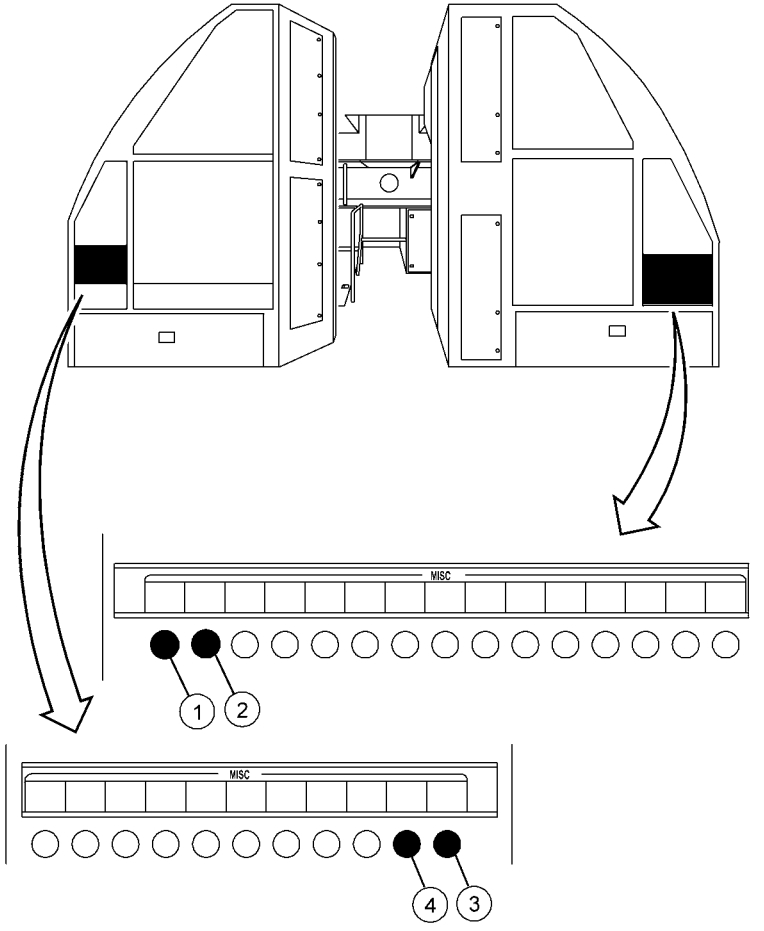
4-1. REPAIR AILERON INTEGRATED FLIGHT CONTROL MODULE BY REPLACING QUAD CONTACT SWITCH.

1. Review ″Section 1 (General Information)″ of this TO for system general warnings, cautions, and notes.

2. Review task ″Functional Input Conditions″ page for task specific safety conditions.

3. Install aileron aircraft ground safety lock (05-10-01, task 01-3).

4. Open **AUX HYDRAULIC PUMP** circuit breakers 1 - 4 on electrical power center and attach warning tag.

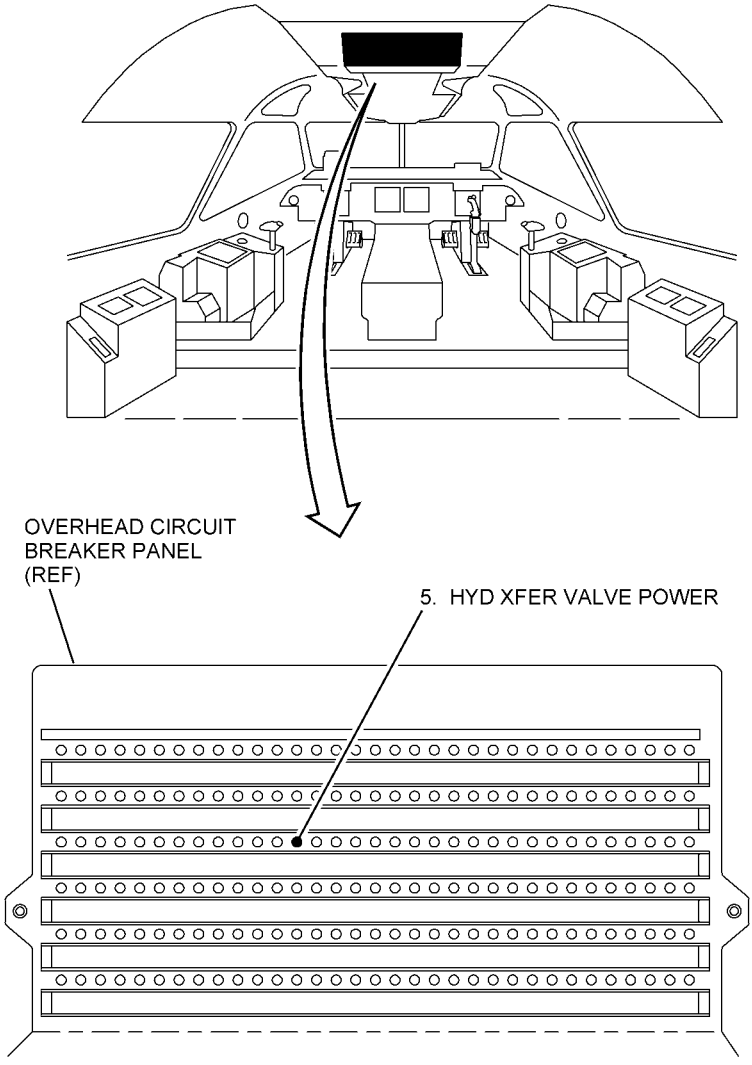


4.

PUMP REF DES	INDEX NO.	CIRCUIT BREAKER	ROW/COLUMN
2921FP001	1	AUX HYDRAULIC PUMP 1	LL/68
2921FP002	2	AUX HYDRAULIC PUMP 2	LL/69
2921FP003	3	AUX HYDRAULIC PUMP 3	LL/11
2921FP004	4	AUX HYDRAULIC PUMP 4	LL/10

ICN-88277-G2713065-001-01

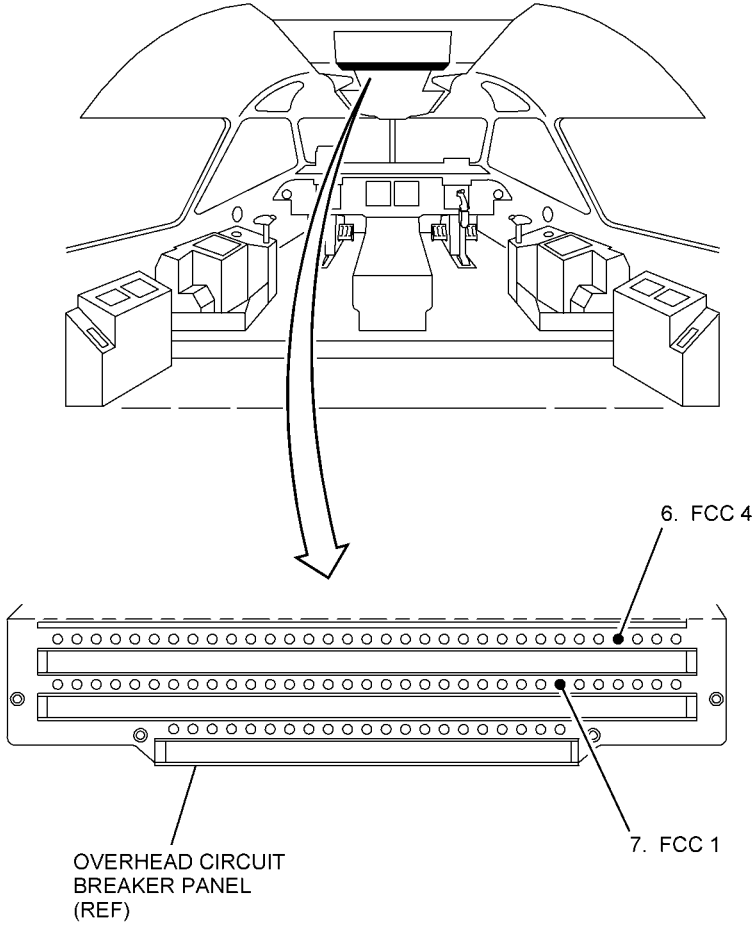
5. Open **HYD XFER VALVE POWER** circuit breaker on overhead circuit breaker panel, row **C**, column **13**, and attach warning tag.



ICN-88277-G2713063-001-01

6. Open **FCC 4** circuit breaker on overhead circuit breaker panel, row **G**, column **30**, and attach warning tag.

7. Open **FCC 1** circuit breaker on overhead circuit breaker panel, row **H**, column **27**, and attach warning tag.



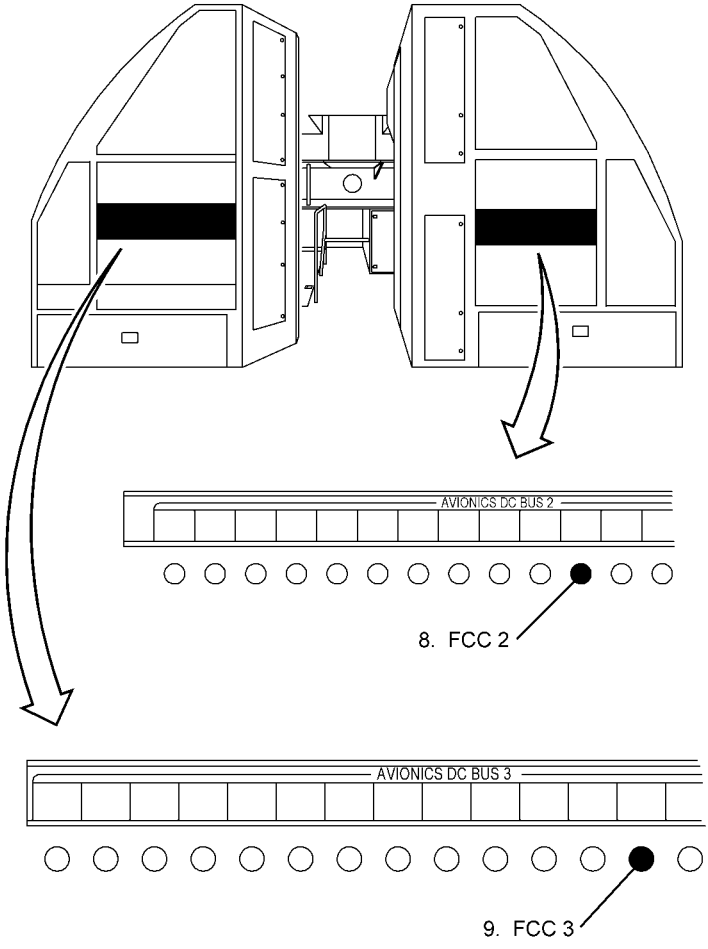
ICN-88277-G2713044-003-01

8. Open **FCC 2** circuit breaker on Electrical Power Center (EPC), row **T**, column **50**, and attach warning tag.

9. Open **FCC 3** circuit breaker on EPC, row **T**, column **24**, and attach warning tag.

10. Lower the lower wing fixed trailing edge door assembly (57-54-01).

IFCM REF DES	DOOR NO.	DOOR REF DES
2713FV001	566BBD	5754CA025
2713FV002	666BBD	5754CA040

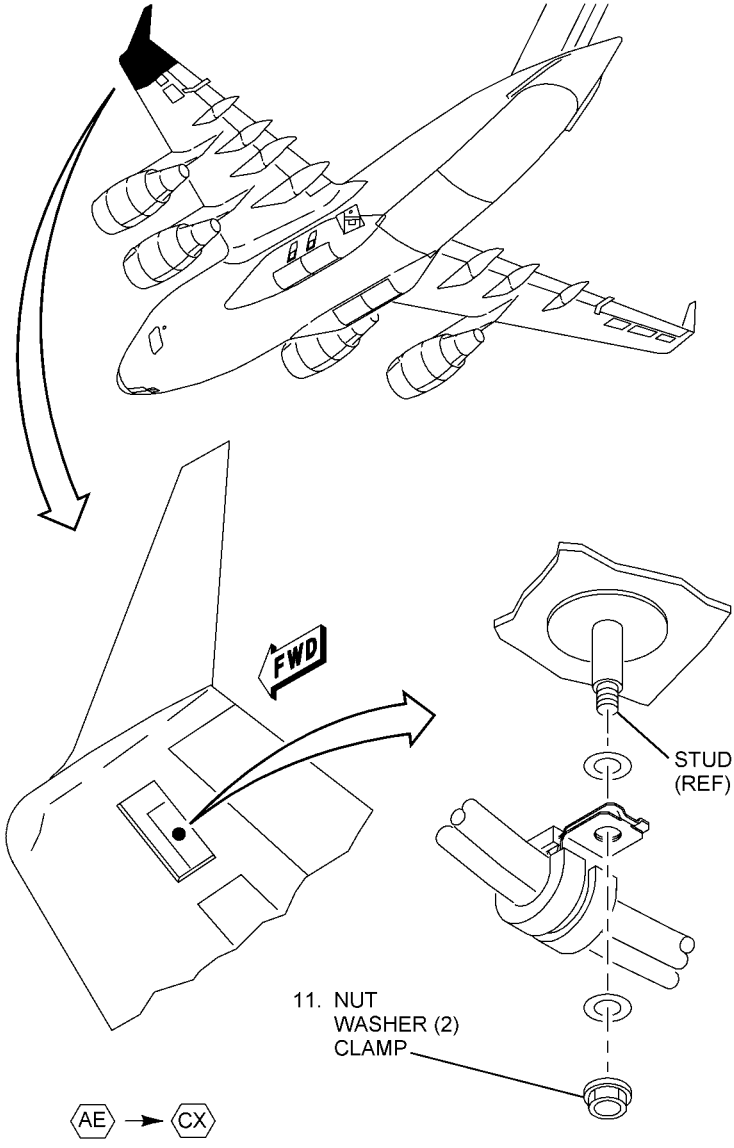


ICN-88277-G2713046-003-01

11. ⟨AA⟩ → ⟨AD⟩ No action required.

11. ⟨AE⟩ → ⟨CX⟩ Remove nut, washers, and clamp from stud.

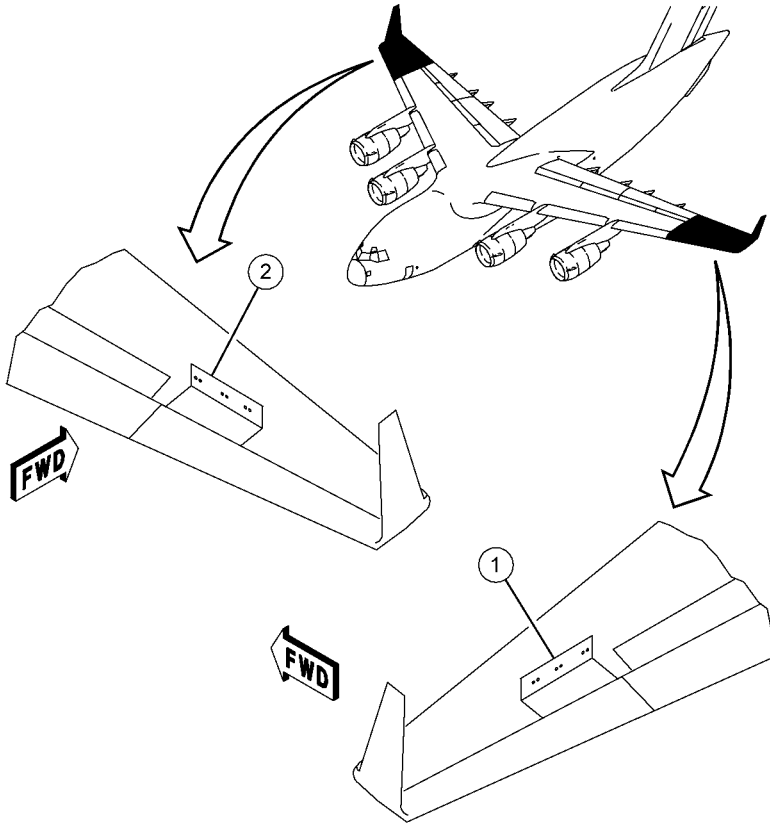
11. ⟨CY⟩ → No action required.



ICN-88277-G2713049-003-01

12. Enter fuselage and wing (00-00-01).

13. Loosen fasteners and open access door.



13.

IFCM REF DES	INDEX NO.	DOOR NO.	FASTENER QTY
2713FV001	1	566ATD	6
2713FV002	2	666ATD	6

ICN-88277-G2713050-004-01

CAUTION

Care shall be maintained when removing or replacing any electrical or mechanical switch on the Integrated Flight Control Module (IFCM). Failure to comply may damage the electrical connector on switch assembly.

NOTE

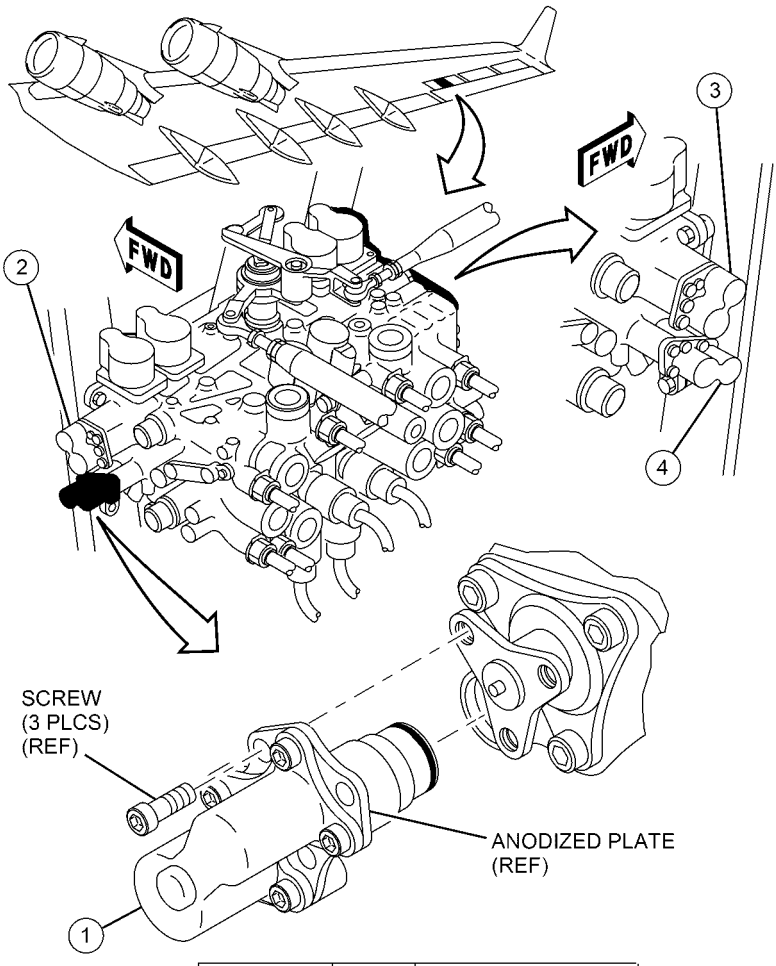
- Electrical quad contact switch index numbers 2 and 4 are for switches which shall be removed and replaced individually as required per the BIT Logic Instruction (BLIN) codes for the left or right aileron IFCM (TO 1300i-2-27FI-00-1, Chapter 3).

- Due to approved software upgrades, post block 19, the Mechanical Quad Electrical Switches are no longer monitored by the on-board Electronic Flight Control System (EFCS). The Mechanical Switches are being removed by the vendor and a cover plate (P/N 328549-3) installed in the mechanical switch locations.

- Mechanical quad contact switch index numbers 1 and 3 are for the two switches in line, looking forward, and which shall be removed and replaced in pairs only as required per the BLIN codes for the left or right aileron IFCM (TO 1300i-2-27FI-00-1, Chapter 3).

- During removal of the switch assembly, note the orientation of safety wire on different sizes of socket head cap screws being removed to aid during installation.

14. Identify and remove safety wire and screws; remove switch cover and anodized plate from quad contact switch.



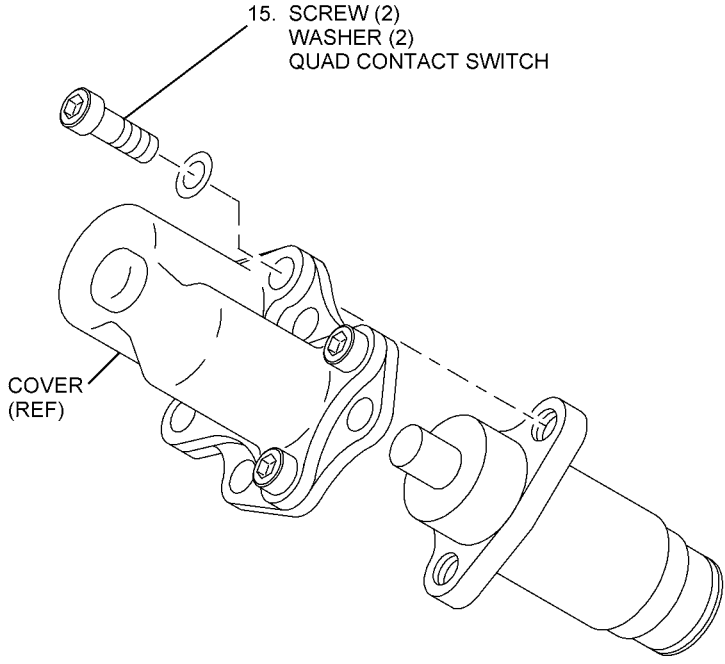
14.

IFCM REF DES	INDEX NO.	QUAD SWITCH
2713FV001 OR 2713FV002	1	MECHANICAL NO. 2
	2	ELECTRICAL NO. 2
	3	MECHANICAL NO. 1
	4	ELECTRICAL NO. 1

(TYPICAL)

ICN-88277-G2713051-004-01

15. Identify and remove screws and washers, and quad contact switch from IFCM manifold.



(TYPICAL)

ICN-88277-G2713052-003-01

16. Remove packing from electrical connector and discard.

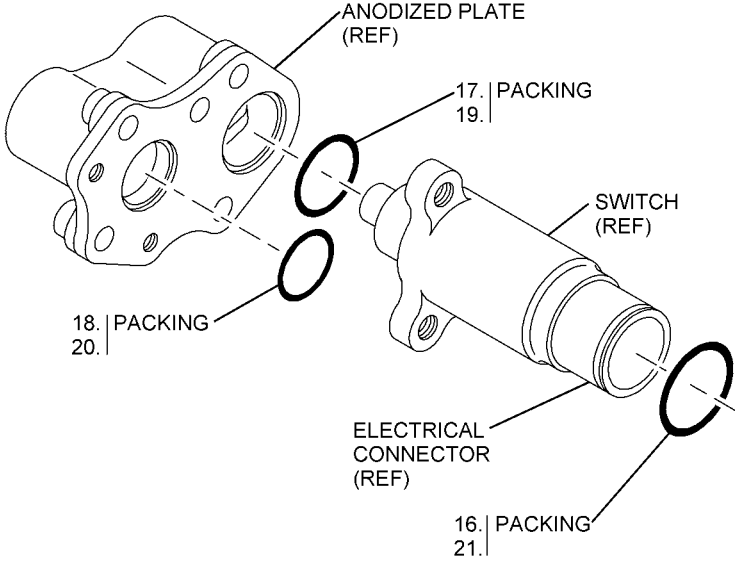
17. Remove packing from switch and discard.

18. Remove packing from face seal opening on the anodized plate and discard.

19. Install packing (PN M83461/1-016) on switch.

20. Install packing (PN M83461/1-015) on face seal opening on the anodized plate.

21. Install packing (PN M83461/1-014) on electrical connector.



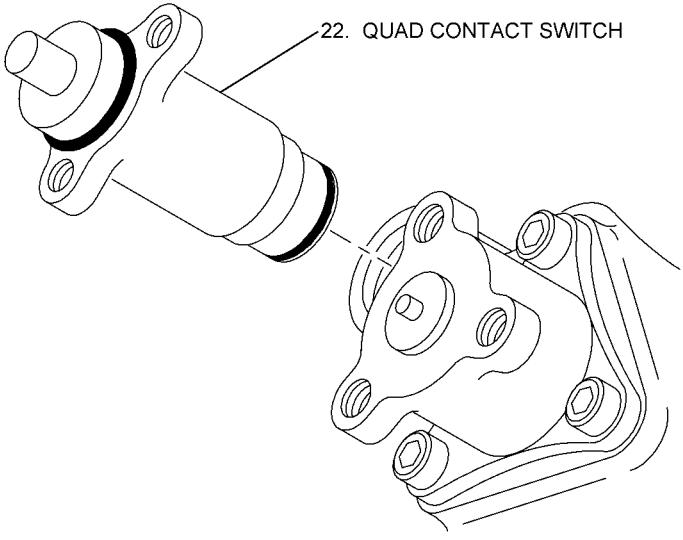
(TYPICAL)

ICN-88277-G2713053-003-01

NOTE

The keyway in the electrical connector of switch assembly shall be aligned with the key of the electrical connector in the IFCM manifold.

22. Loosely install quad contact switch in IFCM manifold.



(TYPICAL)

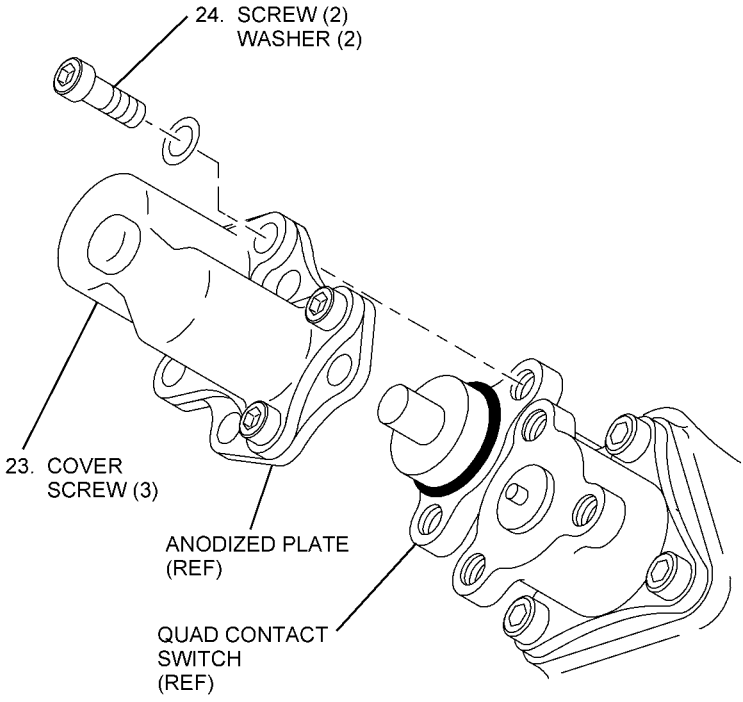
ICN-88277-G2713055-003-01

23. Position cover and anodized plate on switch; install screws and washers as identified; torque **5-7 in-lb**.

24. Install screws as identified and torque **13-16 in-lb**.

25. Secure all screws with safety wire as identified.

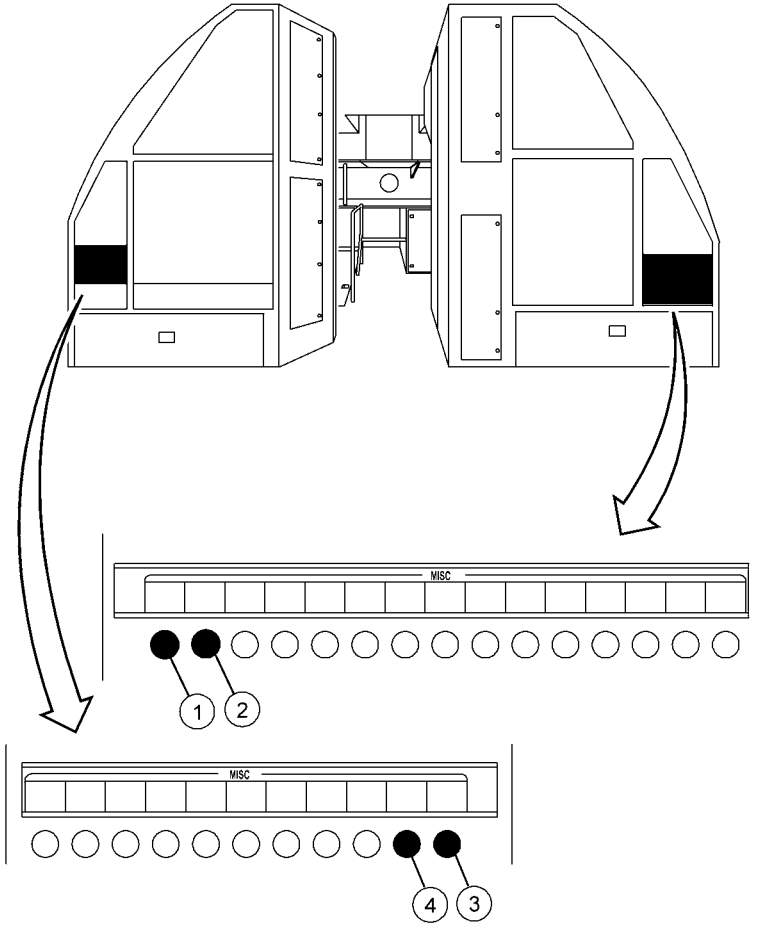
26. Remove aileron aircraft ground safety lock (05-10-01, task 01-4).



(TYPICAL)

ICN-88277-G2713054-003-01

27. Close **AUX HYDRAULIC PUMP** circuit breakers 1 - 4 on electrical power center and remove warning tag.

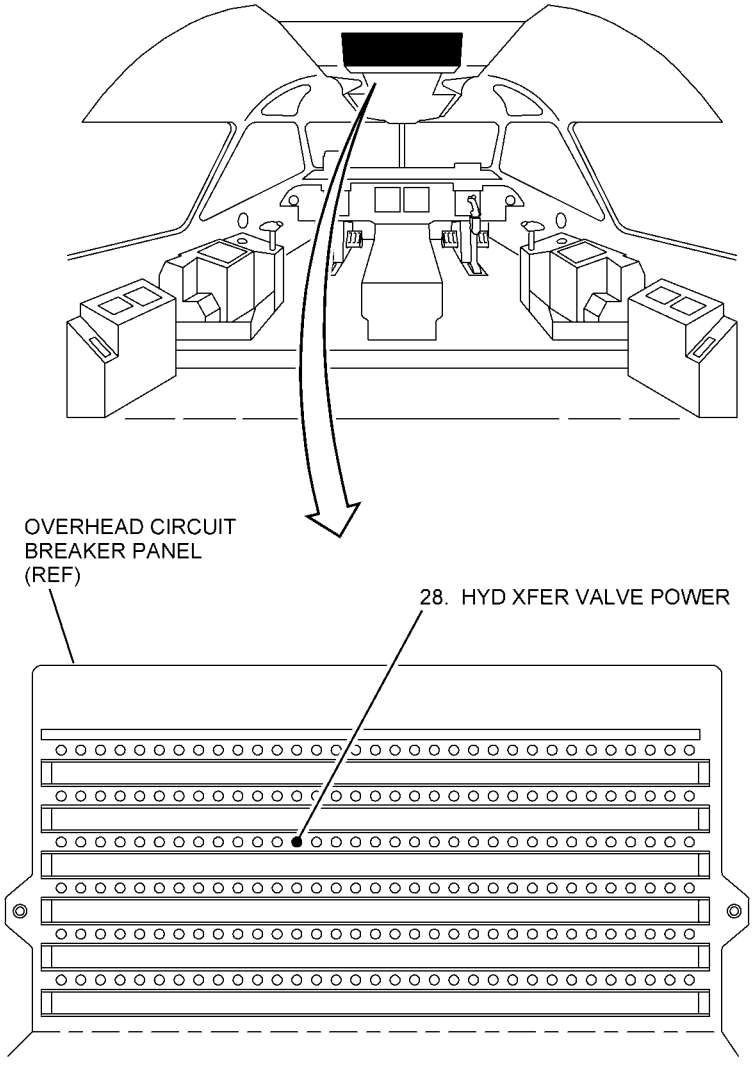


27.

PUMP REF DES	INDEX NO.	CIRCUIT BREAKER	ROW/COLUMN
2921FP001	1	AUX HYDRAULIC PUMP 1	LL/68
2921FP002	2	AUX HYDRAULIC PUMP 2	LL/69
2921FP003	3	AUX HYDRAULIC PUMP 3	LL/11
2921FP004	4	AUX HYDRAULIC PUMP 4	LL/10

ICN-88277-G2713064-001-01

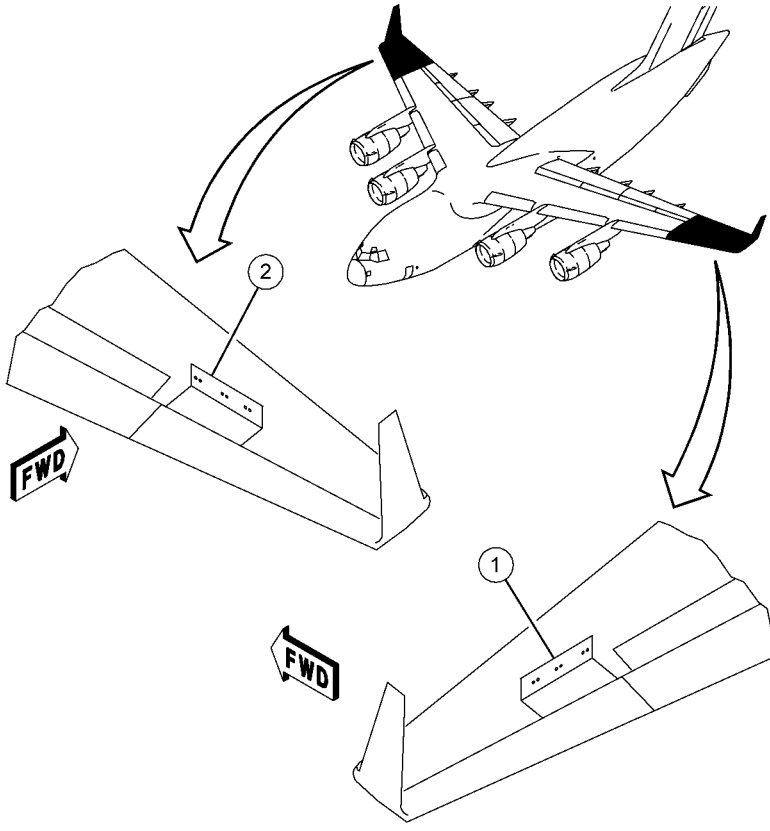
28. Close **HYD XFER VALVE POWER** circuit breaker on overhead circuit breaker panel, row **C**, column **13**, and remove warning tag.



ICN-88277-G2713062-001-01

29. Close access door, and tighten fasteners.

30. Enter fuselage and wing (00-00-01).



29.

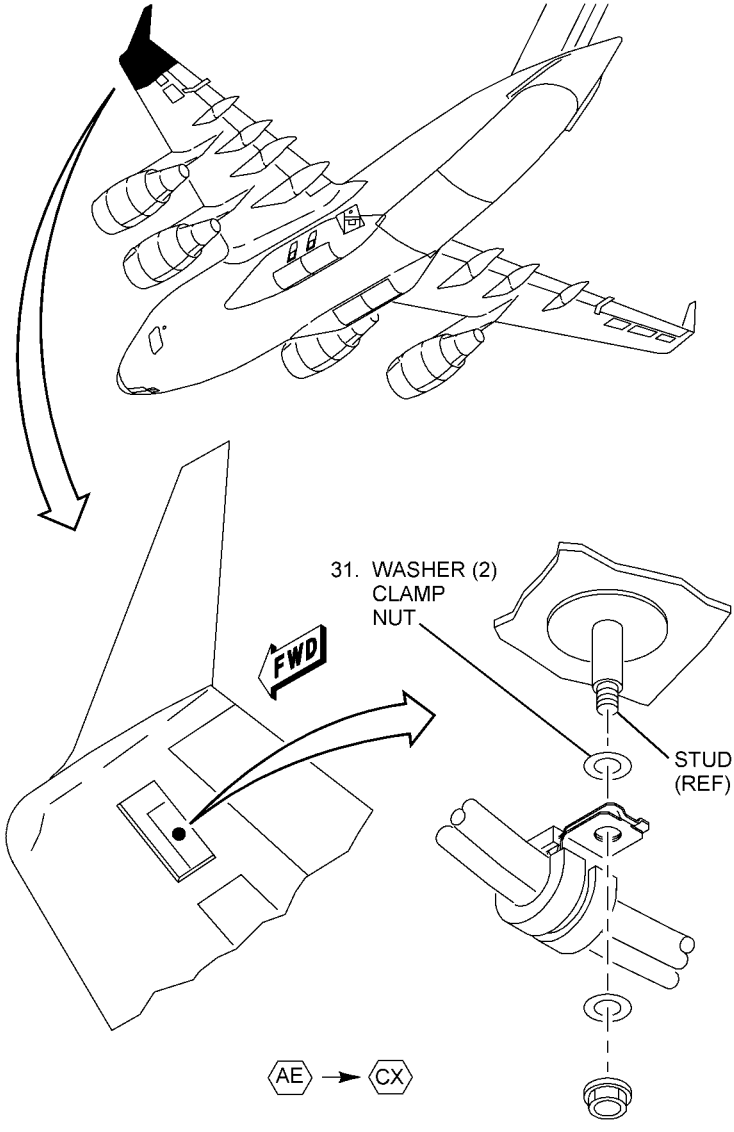
IFCM REF DES	INDEX NO.	DOOR NO.	FASTENER QTY
2713FV001	1	566ATD	6
2713FV002	2	666ATD	6

ICN-88277-G2713056-004-01

31. ⟨AA⟩ → ⟨AD⟩ No action required.

31. ⟨AE⟩ → ⟨CX⟩ Install washer on stud; position clamp, and install washer and nut; tighten nut.

31. ⟨CY⟩ → No action required.



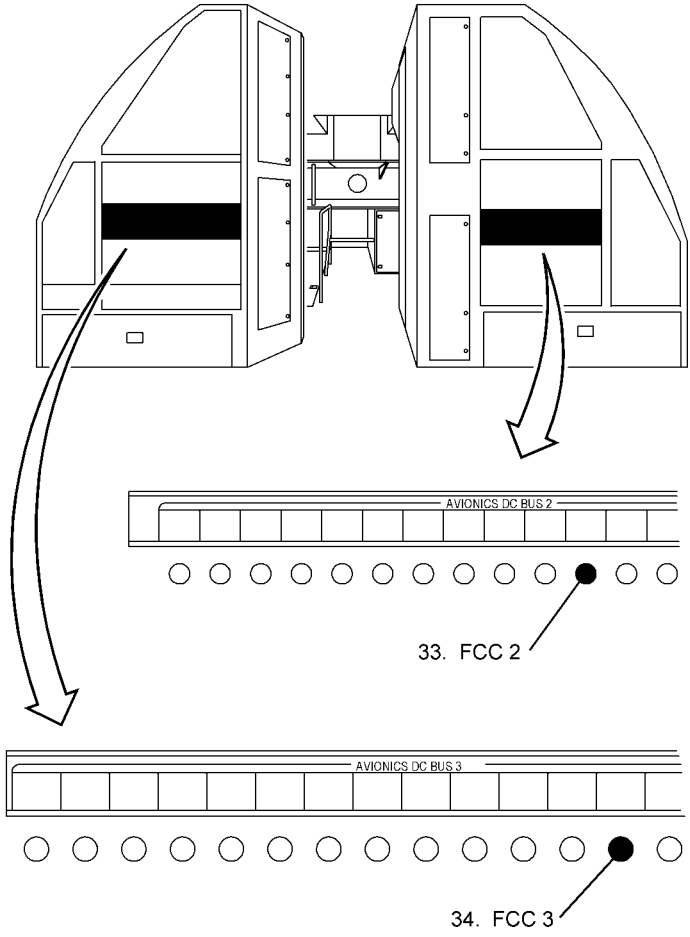
ICN-88277-G2713057-003-01

32. Raise lower wing fixed trailing edge door assembly (57-54-01).

IFCM REF DES	DOOR NO.	DOOR REF DES
2713FV001	566BBD	5754CA025
2713FV002	666BBD	5754CA040

33. Remove warning tag and close **FCC 2** circuit breaker on EPC, row **T**, column **50**.

34. Remove warning tag and close **FCC 3** circuit breaker on EPC, row **T**, column **24**.

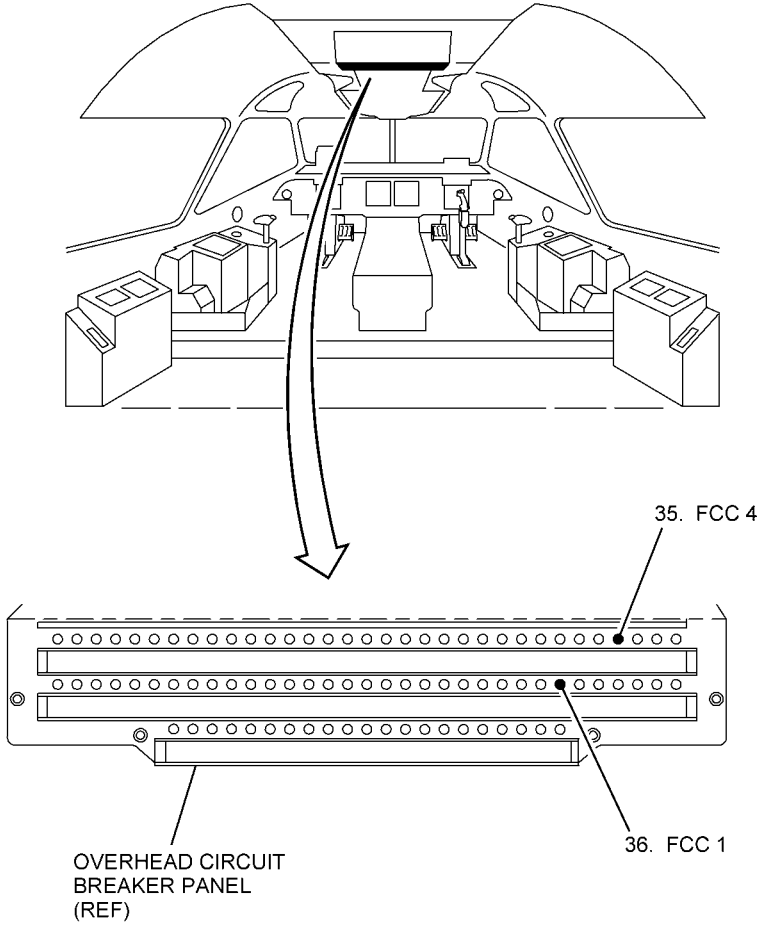


ICN-88277-G2713060-003-01

35. Remove warning tag and close **FCC 4** circuit breaker on overhead circuit breaker panel, row **G**, column **30**.

36. Remove warning tag and close **FCC 1** circuit breaker on overhead circuit breaker panel, row **H**, column **27**.

37. Perform electronic flight control system maintenance built in test (40-00-02, tasks 02-1, 02-2 and/or 02-3, and 02-10).



ICN-88277-G2713061-003-01

AILERON TRIM POSITION INDICATOR (27-14-10)

MASTER INPUT CONDITIONS:

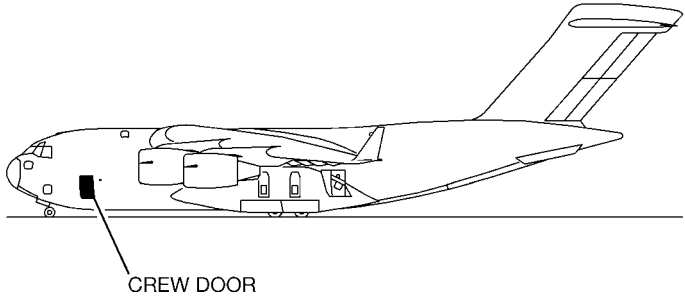
Reference designators:

2714MN001	Aileron Trim Position Indicator

Applicable functions:

-1 Operational Checkout.
-2 Removal.
-3 Installation.

Access data:



ICN-88277-G2714001-002-01

AILERON TRIM POSITION INDICATOR OPERATIONAL CHECKOUT (27-14-10-1)

FUNCTIONAL INPUT CONDITIONS:

Applicability:

	Task
All	All

Additional information:

This procedure consists of the following task:

1-1. Operational checkout.

Additional data:

	Task
TO 1300i-2-10JG-60-1	All
TO 1300i-2-27FI-00-1	All
TO 1300i-2-33FI-00-1	All

Personnel recommended:

	Task
One	All

Safety conditions:

	Task
NA	--

Support equipment:

Nomenclature	PN	Specification	Qty	Task
NA	--	--	--	--

Supplies:

Nomenclature	PN	Specification	Qty	Task
NA	--	--	--	--

1-1. OPERATIONAL CHECKOUT.

1. Review ″Section 1 (General Information)″ of this TO for system general warnings, cautions, and notes.

2. Review task ″Functional Input Conditions″ page for task specific safety conditions.

3. Connect external electrical power (10-61-01, task 01-1).

WARNING

All flight control surfaces shall be clear of personnel and equipment prior to any movement of the surfaces. Failure to comply may cause injury to personnel or damage to aircraft.

4. Press and hold trim switch on control stick to left wing down.

   - **AIL TRIM** indicator moves from 0 to 10 **L WING DN** (27-12-AJ-00).

5. Press and hold trim switch to right wing down.

   - **AIL TRIM** Indicator moves from 10 **L WING DN** to 10 **R WING DN** (27-12-AJ-00).

6. Press and hold trim switch until aileron trim position indicator returns to 0.