TO 1300i-2-27JG-80-2

TECHNICAL MANUAL

JOB GUIDE
ORGANIZATIONAL MAINTENANCE

FLIGHT CONTROLS SLATS
(27-80-00 AND 27-80-03 THROUGH 27-81-10)

USAF SERIES
300i
AIRCRAFT

MCDONNELL DOUGLAS CORPORATION
MILITARY TRANSPORT AIRCRAFT
F33657-81-C-2108
FA8526-21-D-0001

THIS MANUAL SUPERSEDES TO 1300i-2-27JG-80-2 DATED 1 AUGUST 2020.

Basic and all changes have been merged to make this a complete publication.
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T-2

1 OCTOBER 2022

INSERT LATEST CHANGED PAGES. DESTROY SUPERSEDED PAGES.

LIST OF EFFECTIVE PAGES

NOTE: The portion of the text affected by the changes is indicated by a vertical line in the outer margins of the page. Changes to illustrations are indicated by miniature pointing hands.

Dates of issue for original and changed pages are:

Original . . . . . 0 . . . . . 1 Oct 22

TOTAL NUMBER OF PAGES IN THIS PUBLICATION IS 194 CONSISTING OF THE FOLLOWING:

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TABLE OF CONTENTS

INTRODUCTION

SCOPE.

This manual contains maintenance procedures for the bleed, removal, installation, and repair of slats system components.

MODEL(S) COVERED.

All

ABBREVIATIONS.

The following is a list of non-standard abbreviations used throughout this manual:

EPC	Electrical Power Center
HSR	Hydraulic System Reservoir
PLCS	Places
SDS	Safety Data Sheet

CHANGE REQUEST.

Recommended changes to this manual shall be submitted in accordance with TO 00-5-1.300i

TO INFORMATION.

General 300i TO/eTO, TO Manager, Supplement and finalized Recommended Change (RC) information can be found in the Enhanced Technical Information Management System (ETIMS), System of Record.

LIST OF TIME COMPLIANCE TECHNICAL ORDERS (TCTO).

This list of TCTO's contains all current TCTO's that affect the technical content of text or illustrations found in this manual.

TCTO NUMBER	TITLE	TCTO DATE	APPLICABILITY

SECTION 1

GENERAL INFORMATION (27-80-00)

1-1. GENERAL INFORMATION.

1-2. This section provides general information that is essential for ensuring complete and safe maintenance procedures contained throughout this manual.

1-3. When operating auxiliary pump below 15 degrees Fahrenheit a 30 seconds on/30 seconds off duty cycle for a maximum 10 cycles may be required to reach full hydraulic pressure of 3800 to 4200 psi. Allow 10 minutes for cooling and repeat cycle.

1-4. Hydraulic system No. 2 may require 45 seconds before reaching full hydraulic pressure of 3800 to 4200 psi.

1-5. All Hydraulic System Reservoir (HSR) servicing levels specified in this manual require the aircraft to be in parked configuration.

1-6. Prior to reading reservoir levels for systems No. 1 and No. 4, verify hydraulic system air content is within 3 quarts difference, system No. 2 is within 5 quarts difference, system No. 3 is within 4 quarts difference, pressurized to unpressurized (12-29-03).

1-7. Flight control surfaces are to be cleared prior to turning off hydraulic auxiliary pumps from loadmaster control panels. Flight control surface movement may occur.

1-8. All adhesive sealants, sealants, and compounds used in this manual are listed with a primary part number and/or primary specification number. Any suitable substitutes and/or interchangeable adhesive sealants, sealants, and compounds may be used unless otherwise specified. Suitable substitutes and/or interchangeable adhesive sealants, sealants, and compounds are listed in the system peculiar corrosion control manual (TO 1300i-23, Chapter 1, Section III).

1-8A. Proper tube clearances shall be maintained after hydraulic component replacement, where loosening of tube support clamps and/or brackets on interfacing hydraulic lines is necessary to relieve tube-to-fitting insertion and facilitate component removal. Minimum allowable hydraulic tubing clearances may be found in the system's General System (GS) manual (Refer to TO 1300i-2-29GS-00-1, Chapter 2).

1-9. GENERAL WARNINGS, CAUTIONS, AND NOTES.

WARNING

- All flight control surfaces and thrust reversers shall be clear of personnel and equipment prior to applying or removing hydraulic power. Failure to comply may cause injury to personnel and damage to aircraft.

- All flight control surfaces and thrust reversers shall be clear of personnel and equipment prior to movement of any surfaces. Failure to comply may cause injury to personnel and damage to aircraft.

- Warning tags shall be attached to all opened circuit breakers as directed by the technical order and an applicable warning statement shall be entered in the AFTO Form/IMT 781A, IAW TO 00-20-1, AFI 21-101 and TO 1300i-2-00GV-00-1, Chapter 5 anytime exiting cockpit circuit breaker area before task completion or there is a delay in maintenance where task cannot be fully accomplished. Failure to comply may cause injury to personnel or damage to aircraft.

CAUTION

Air in a hydraulic system can cause numerous malfunctions, from a total system failure to a minor indication problem. If you suspect air has been inducted into a system by removing a hydraulic component or a line, refer to the hydraulic system bleed procedure (12-29-08). Failure to comply may cause damage to aircraft.

SECTION 2

ALTERNATE SLAT ACTUATION SYSTEM BLEED (27-80-03)

GENERAL MAINTENANCE INPUT CONDITIONS:

Applicability:

	Task
All	All

Additional information:

This procedure consists of the following tasks:

03-1. Preparation.
03-2. Alternate slat actuation bleed.
03-3. Follow-on maintenance.

Additional data:

	Task
TO 1300i-2-29GS-00-1	All
TO 1300i-2-10JG-10-1	03-1, 03-3
TO 1300i-2-10JG-60-1	03-1
TO 1300i-2-12JG-29-1	03-1, 03-3
TO 1300i-2-23JG-40-1	03-1, 03-3
TO 1300i-23	03-2

Personnel recommended:

	Task
Three	03-2

	Task
Four	03-1, 03-3

Person (A) performs task.

Person (B) assists person (A).

Person (C) performs flight station/aircraft operations.

Person (D) operates hydraulic test stand.

Safety conditions:

	Task
WARNING	
• Hydraulic actuated components, including flight control surfaces, must be clear of personnel and support equipment prior to applying hydraulic pressure or flow. Components and surfaces may move with pressure application. Failure to comply may cause injury to personnel and damage to equipment.	All
• Gust damper accumulator system maintains hydraulic pressure on systems 2 and 3 to prevent wind induced movement and subsequent control surface damage. As a result of this maintained system pressure, maintenance personnel shall relieve accumulator precharge pressure prior to performing maintenance on systems 2 and 3 in accordance with TO 1300i-2-29GS-00-1, Chapter 2. Failure to comply may cause injury to personnel and damage to equipment.	All

Task

WARNING - Continued

- The use of personal protective equipment is mandatory to perform this procedure. The applicable Safety Data Sheet (SDS) will identify special protection information. Failure to comply may cause injury to personnel. — All

CAUTION

- Hydraulic test stand must be serviced with MIL-H-83282 hydraulic fluid. Using a test stand with other than MIL-H-83282 fluid may result in system contamination and failure. Failure to comply may cause damage to equipment. — All

- Hydraulic test stand hoses must be used in specific pairs. Test stands have two pressure systems with specific pressure and return hoses for each system. Mixing hoses between systems may cause hose or aircraft component failure or equipment damage. — All

Support equipment:

Nomenclature	PN	Specification	Qty	Task
Hose Assembly, Hydraulic	17G010063-1	--	1	03-1
Hose Assembly, Hydraulic	17G010064-1	--	1	03-1
Pail, Utility	B12R	--	AR	03-2
Test Stand, Hydraulic	87025-100	--	1	03-1
Test Stand, Hydraulic	87025-100 MOD	--	1	03-1
Test Stand, Hydraulic	88043-100	--	1	03-1
Test Stand, Hydraulic	88043-1001	--	1	03-1
Test Stand, Hydraulic	9780-0095	--	1	03-1
Wrench, Torque	--	(0-150 in-lb)	1	03-2

Supplies:

Nomenclature	PN	Specification	Qty	Task
Sealant	PR-1775 B-2	AMS 3265	AR	03-2
Sealant	PR-1775 B-2	MIL-PRF-81733	AR	03-2
Tag, Warning	--	--	6	03-1

03-1. PREPARATION.

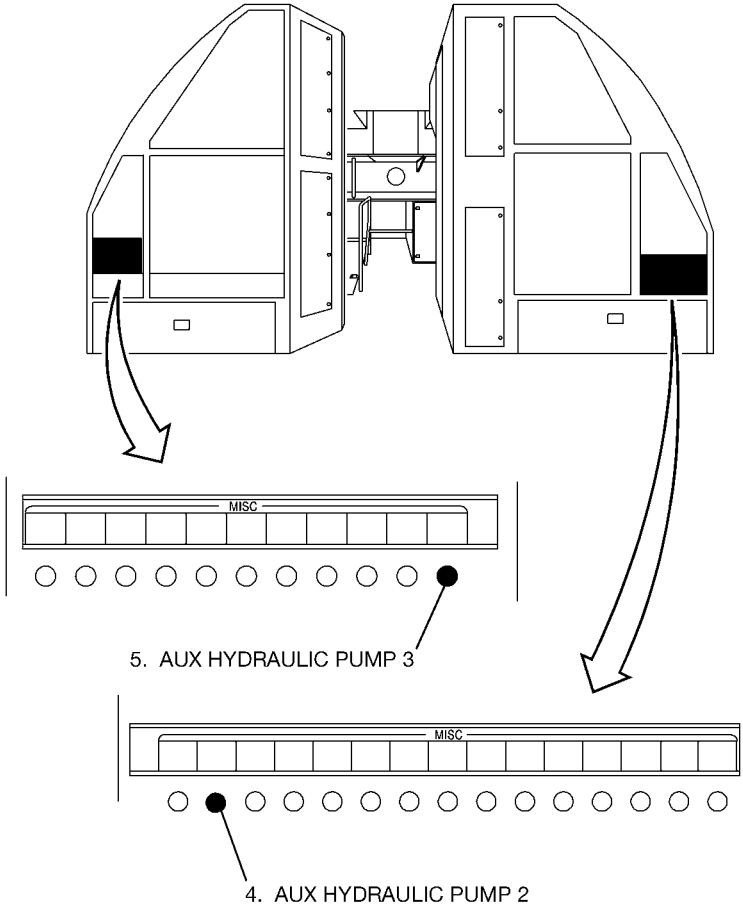
1. Review ″Section 1 (General Information)″ of this TO for system general warnings, cautions, and notes.

2. Review task ″General Maintenance Input Conditions″ page for task specific safety conditions.

3. Connect external electrical power (10-61-01, task 01-1).

4. (C) Open **AUX HYDRAULIC PUMP 2** circuit breaker on Electrical Power Center (EPC), row **LL**, column **69**, and attach warning tag.

5. (C) Open **AUX HYDRAULIC PUMP 3** circuit breaker on EPC, row **LL**, column **11**, and attach warning tag.



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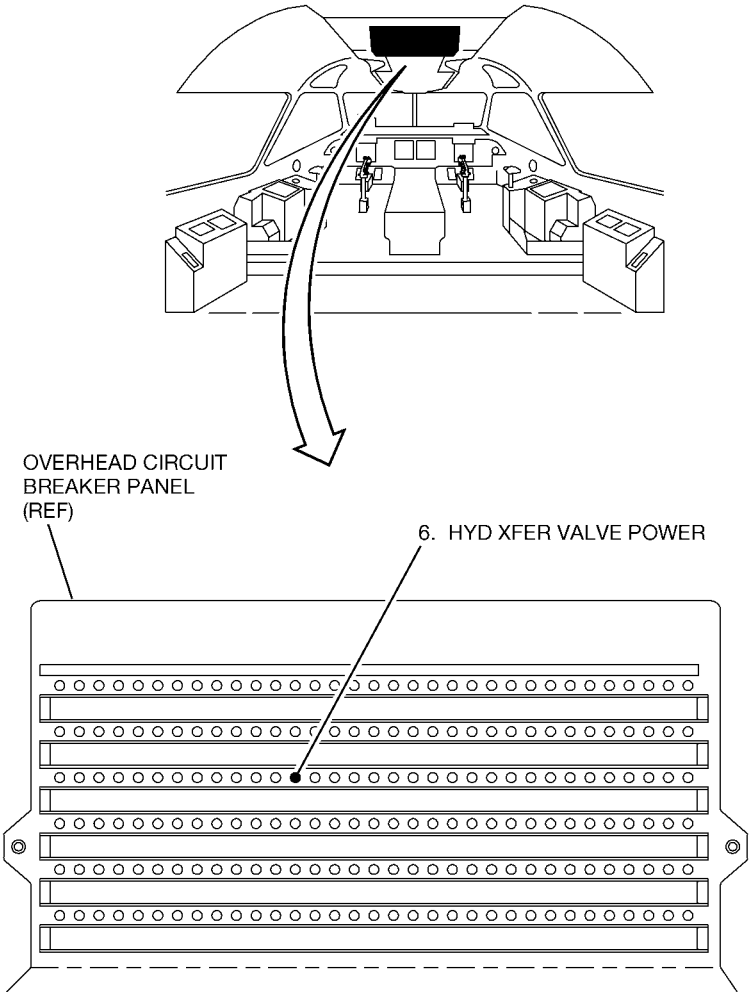
6. (C) Open **HYD XFER VALVE POWER** circuit breaker on overhead circuit breaker panel, row **C**, column **13**, and attach warning tag.



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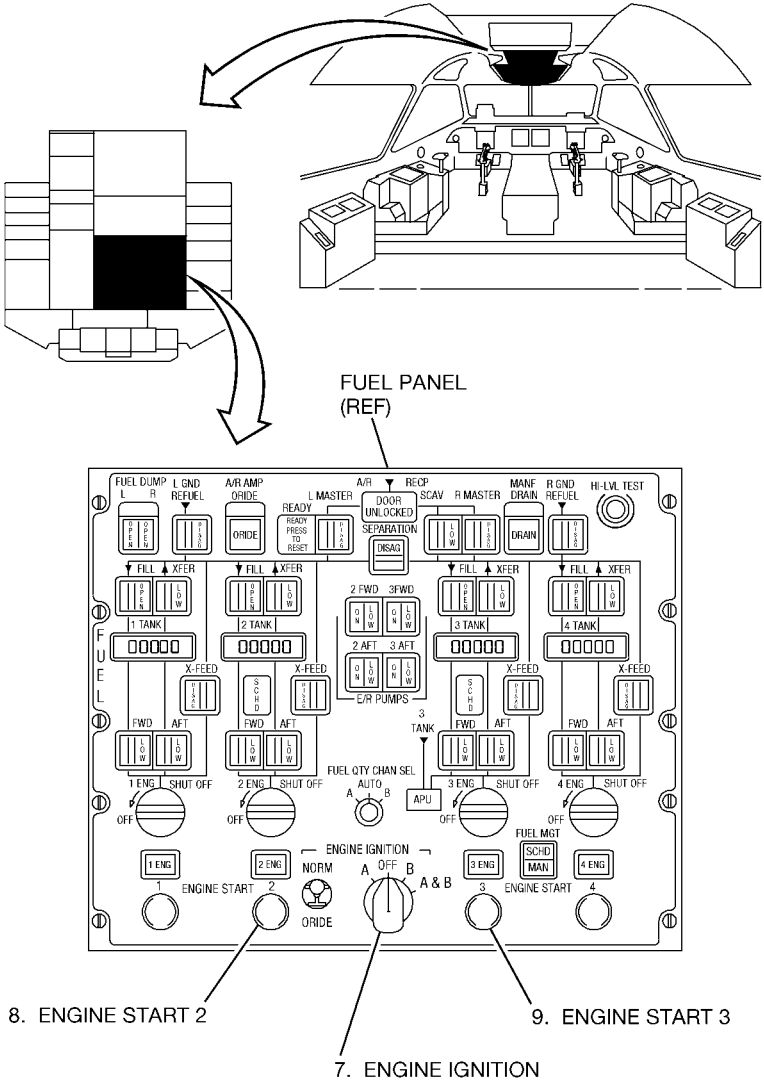
7. (C) Ensure **ENGINE IGNITION** switch on **FUEL** panel is **OFF** and attach warning tag.

8. (C) Attach warning tag to **ENGINE START 2** switch.

9. (C) Attach warning tag to **ENGINE START 3** switch.



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CAUTION

When winds are above 14 knots, aircraft ground safety locks shall be installed per 10-10-00, para 1-12 prior to bleeding down the gust damper system(s). Failure to comply may cause damage to control surfaces.

10. (C) Open access door (265TZM(AT)).

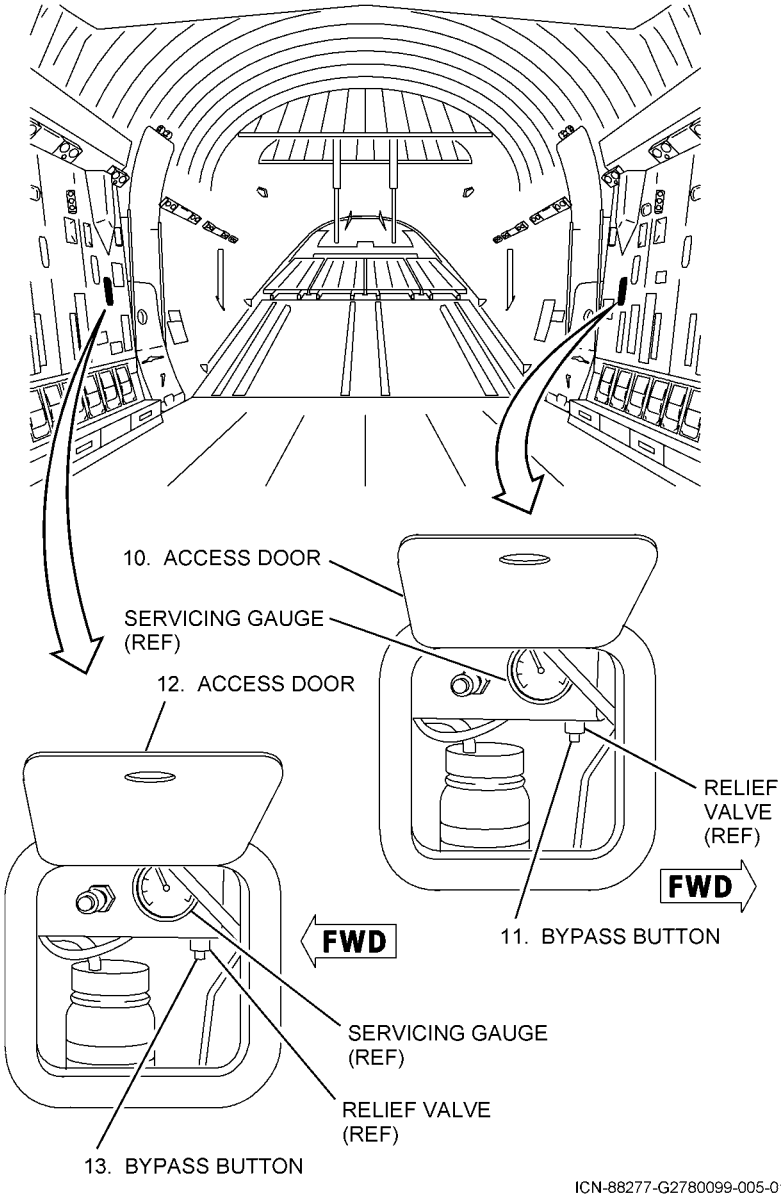
11. (C) Press bypass button on hydraulic system 2 relief valve until servicing gauge stops decreasing.

    - Servicing gauge reads nitrogen precharge (TO 1300i-2-12JG-29-1, 12-29-00, para 1-21).

12. (C) Open access door (266TZM(AT)).

13. (C) Press bypass button on hydraulic system 3 relief valve until servicing gauge stops decreasing.

    - Servicing gauge reads nitrogen precharge (TO 1300i-2-12JG-29-1, 12-29-00, para 1-21).



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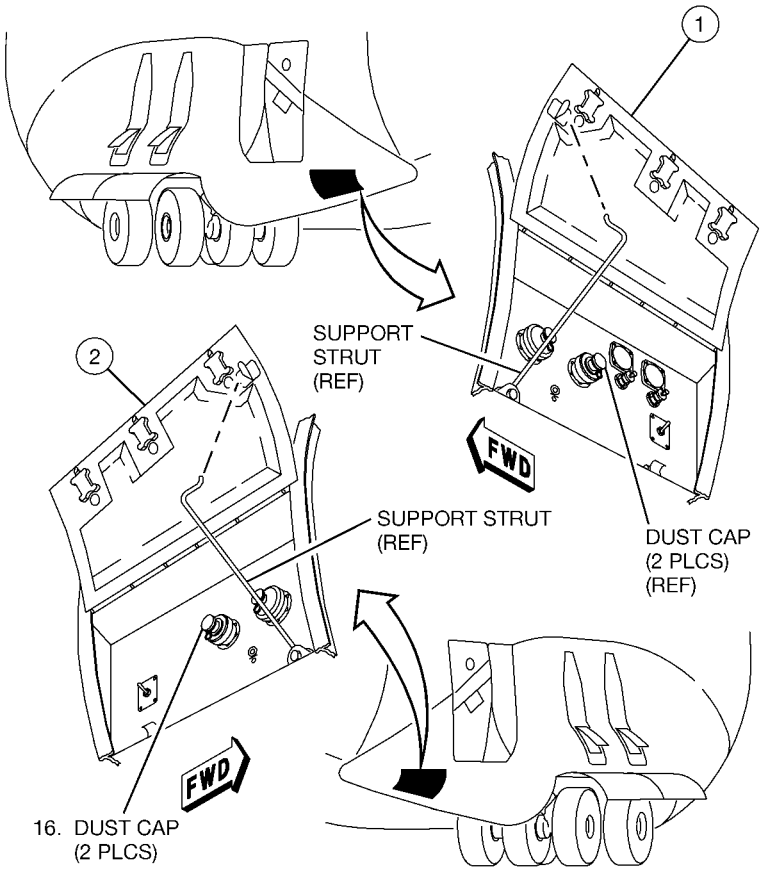
14. Perform maintenance interphone operation (23-41-02, task 02-3).

15. (A,B) Unlatch and open doors; install support struts.

16. (A,B) Remove dust caps.



15.

INDEX NO.	HYDRAULIC SYSTEM	DOOR NO.	LATCH QTY
1	2	173CLD	3
2	3	183CRD	3

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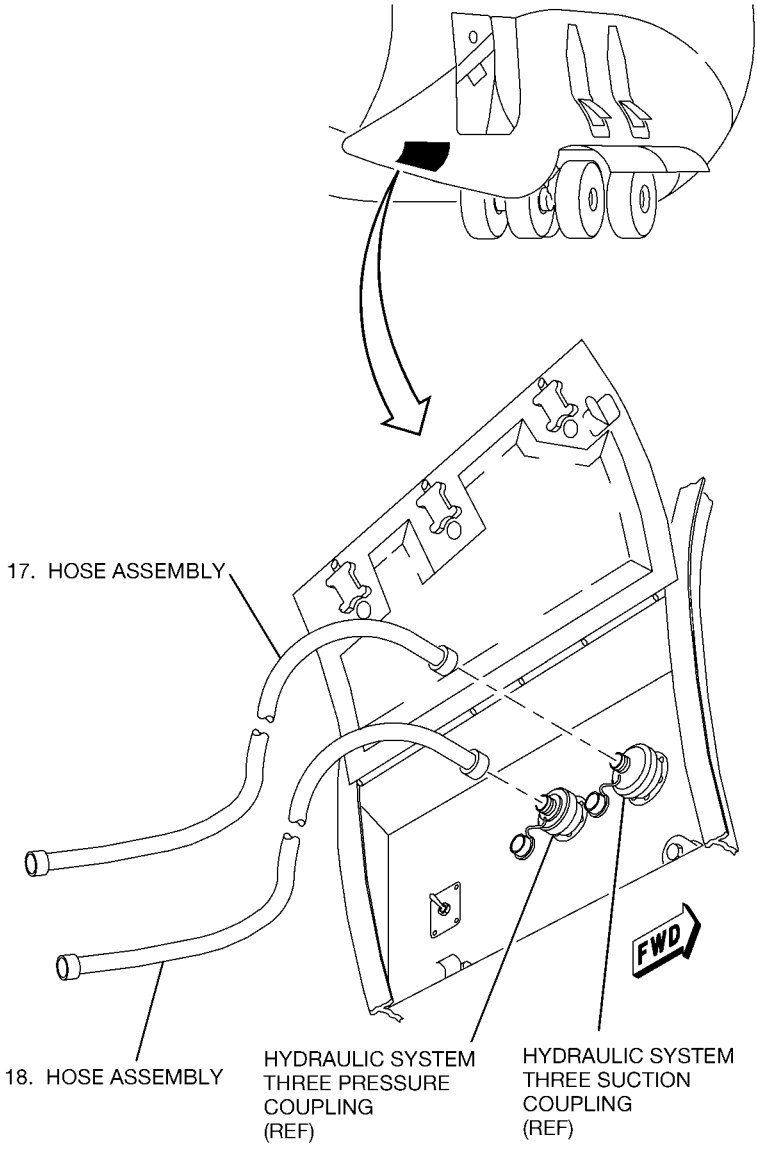
17. (A,B) Connect hose assembly (PN 17G010064-1) to hydraulic system three suction coupling.

18. (A,B) Connect hose assembly (PN 17G010063-1) to hydraulic system three pressure coupling.



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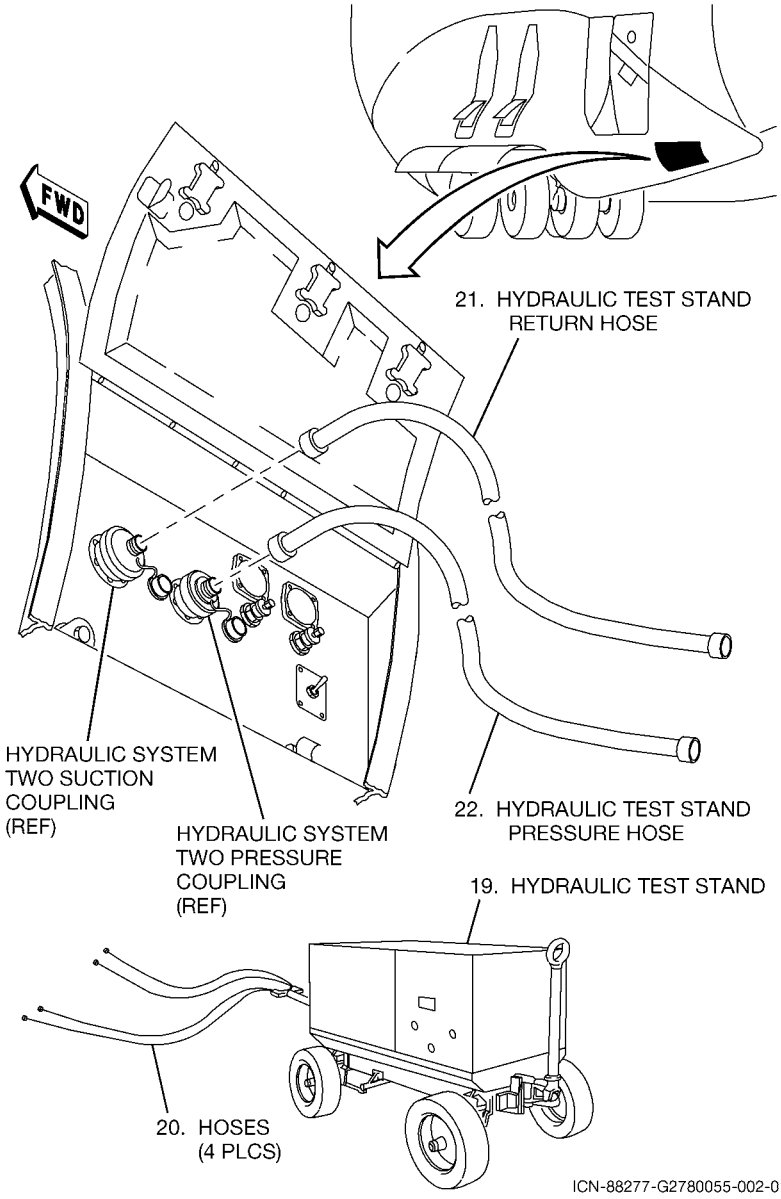
19. (A,B) Position hydraulic test stand next to left aft main landing gear pod.

20. (A,B) Extend test stand pressure and return hoses from test stand.

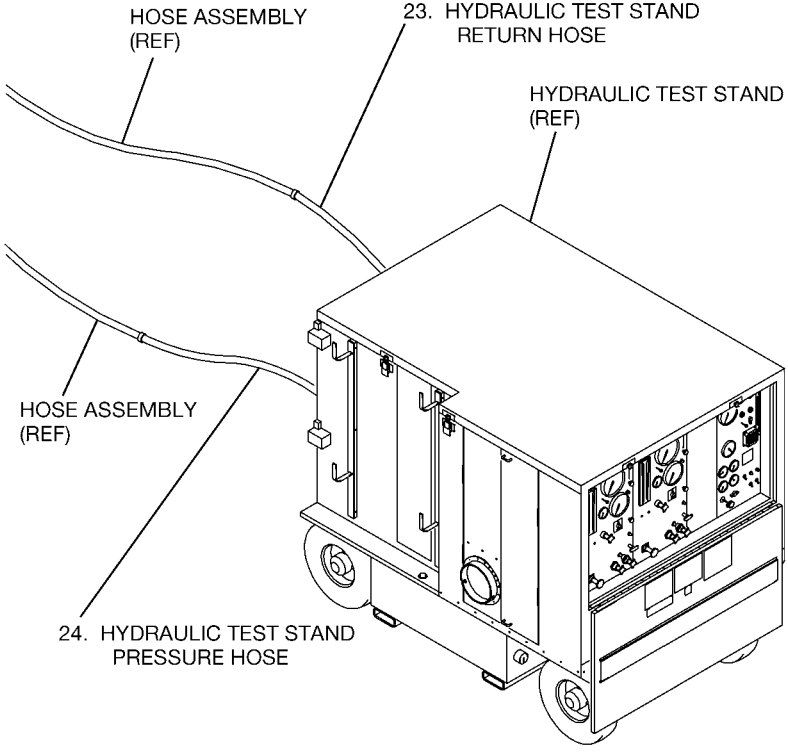
21. (A,B) Connect hydraulic test stand return hose to hydraulic system two suction coupling.

22. (A,B) Connect hydraulic test stand pressure hose to hydraulic system two pressure coupling.


FWD
21. HYDRAULIC TEST STAND RETURN HOSE
HYDRAULIC SYSTEM TWO SUCTION COUPLING (REF)
HYDRAULIC SYSTEM TWO PRESSURE COUPLING (REF)
22. HYDRAULIC TEST STAND PRESSURE HOSE
19. HYDRAULIC TEST STAND
20. HOSES (4 PLCS)
ICN-88277-G2780055-002-01

23. (A,B) Connect hydraulic test stand return hose to hose assembly (PN 17G010064-1).

24. (A,B) Connect hydraulic test stand pressure hose to hose assembly (PN 17G010063-1).



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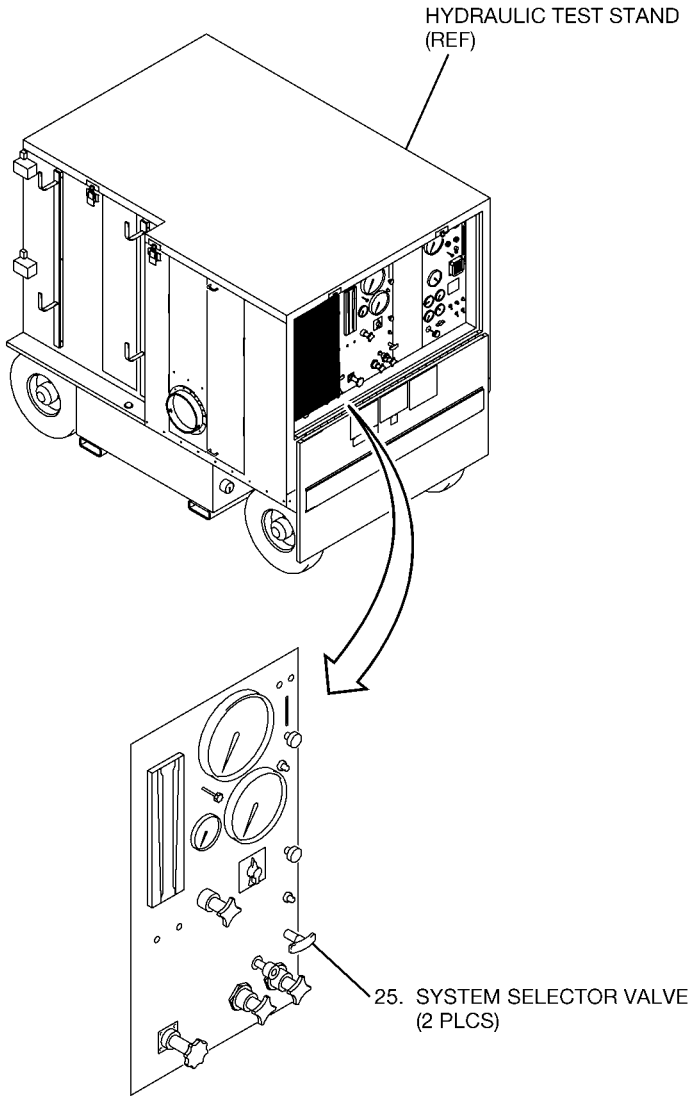
NOTE

On hydraulic test stand PN 87025-100 and 87025-100 MOD ensure **RESERVOIR SELECTOR** valves are set to **AIRCRAFT**.

25. (A,B) Ensure **SYSTEM SELECTOR** valves on hydraulic test stand are set to **AIRCRAFT RESERVOIR**.



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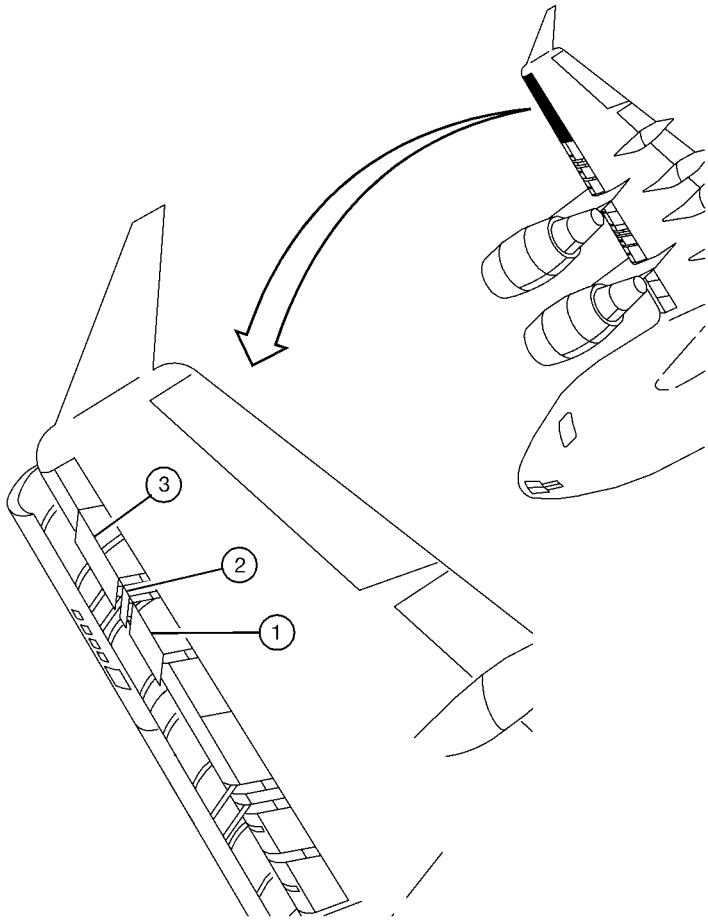
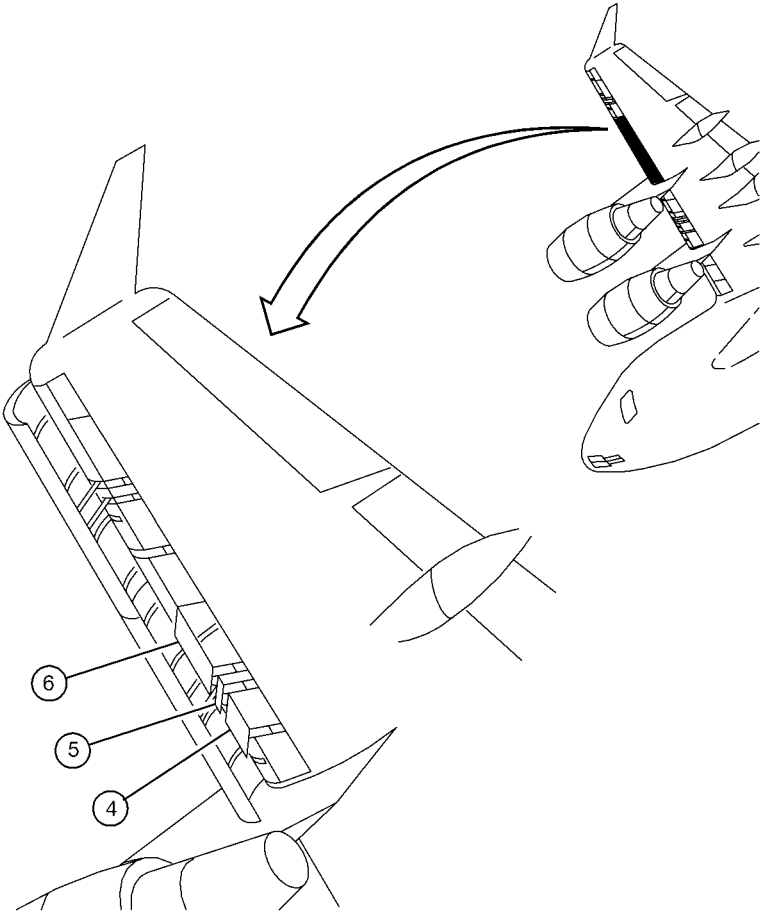
26. (A,B) Loosen fasteners; unlatch and open leading edge lower access doors.



26

INDEX NO.	DOOR NO.	FASTENER QTY	LATCH QTY
1	533JBD	-	4
2	533KBD	-	2
3	533LBD	-	5

ICN-88277-G2780058-002-01



26

INDEX NO.	DOOR NO.	FASTENER QTY	LATCH QTY
4	533EBD	4	4
5	533FBD	2	2
6	533GBD	-	5

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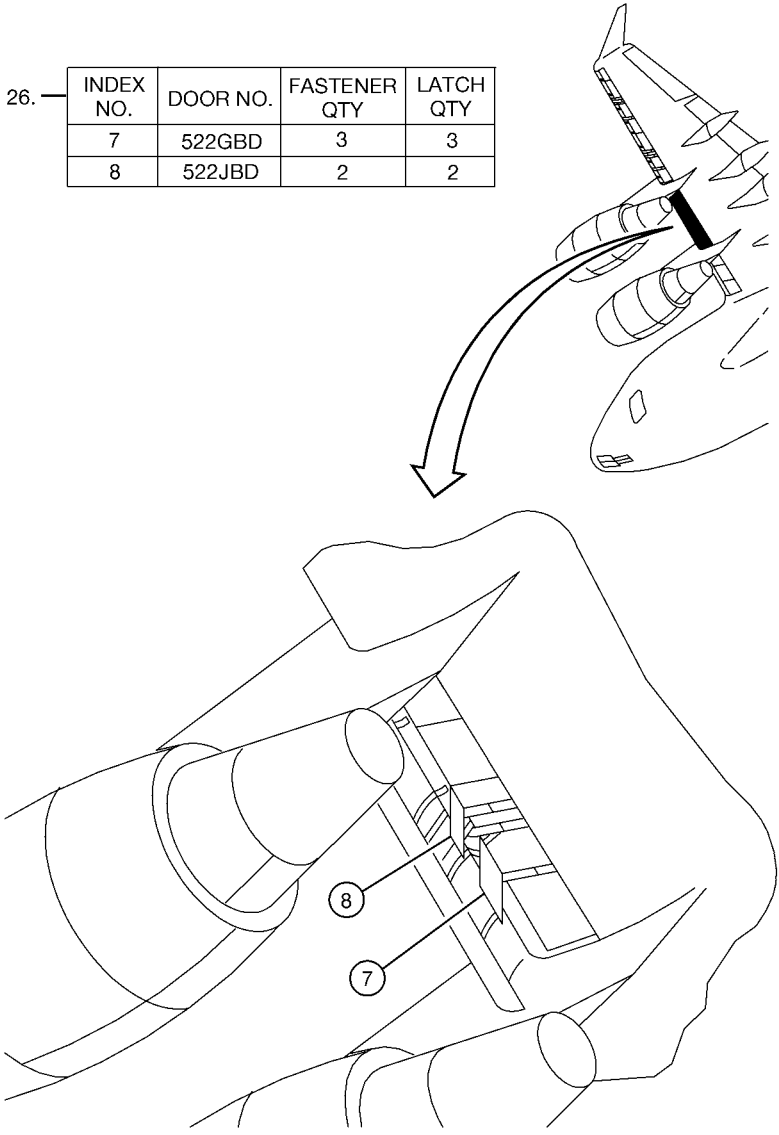
26.

INDEX NO.	DOOR NO.	FASTENER QTY	LATCH QTY
7	522GBD	3	3
8	522JBD	2	2



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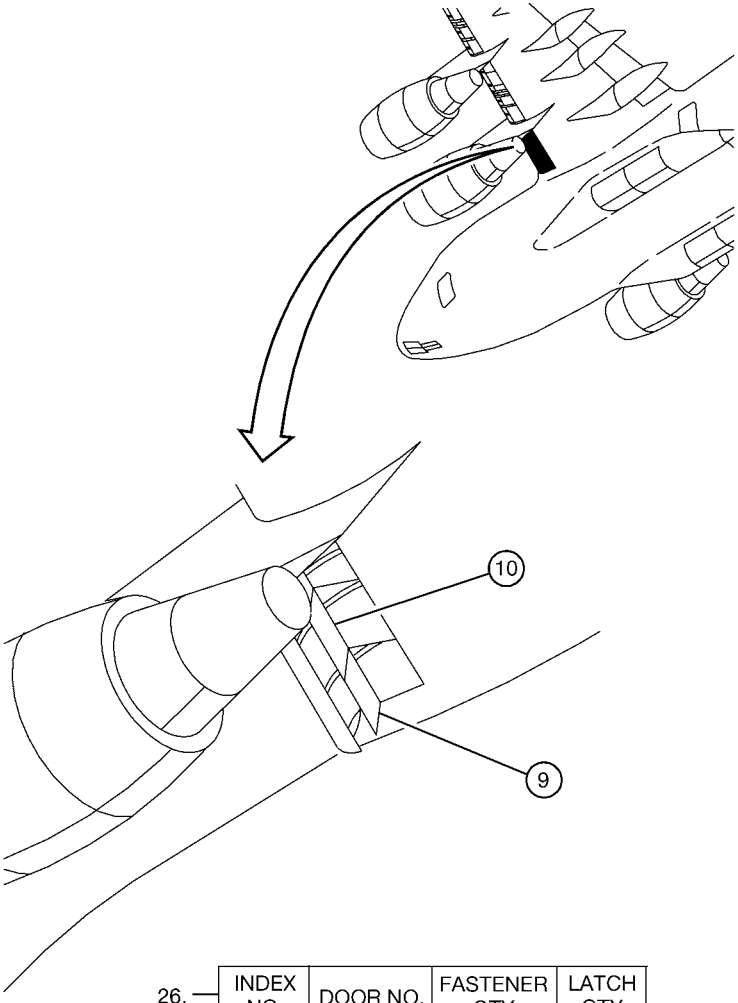

26.

INDEX NO.	DOOR NO.	FASTENER QTY	LATCH QTY
9	512ABD	5	5
10	512BBD	5	5

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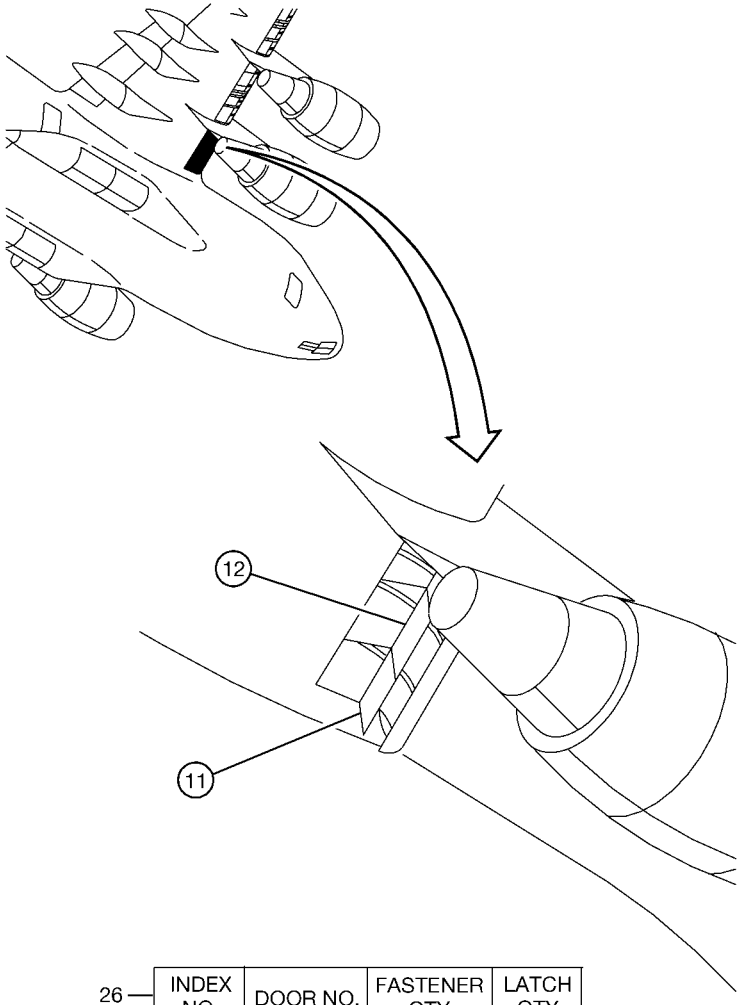


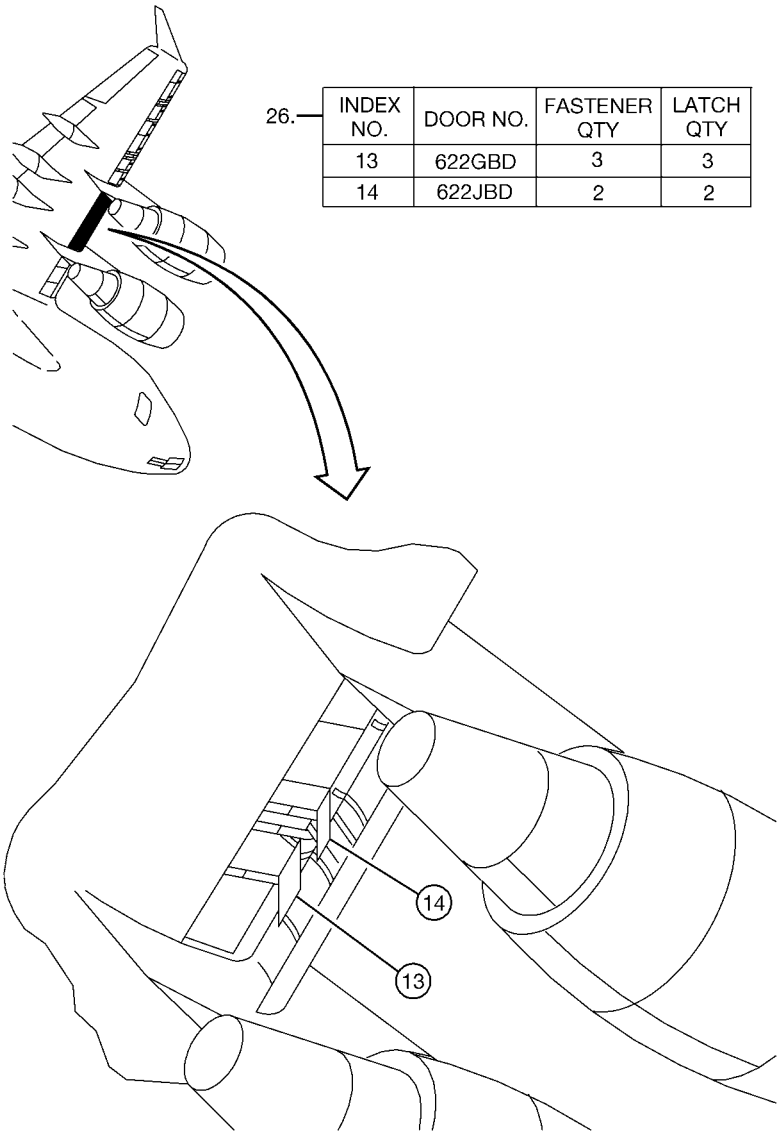
26

INDEX NO.	DOOR NO.	FASTENER QTY	LATCH QTY
11	612ABD	5	5
12	612BBD	5	5

ICN-88277-G2780062-002-01

26.

INDEX NO.	DOOR NO.	FASTENER QTY	LATCH QTY
13	622GBD	3	3
14	622JBD	2	2



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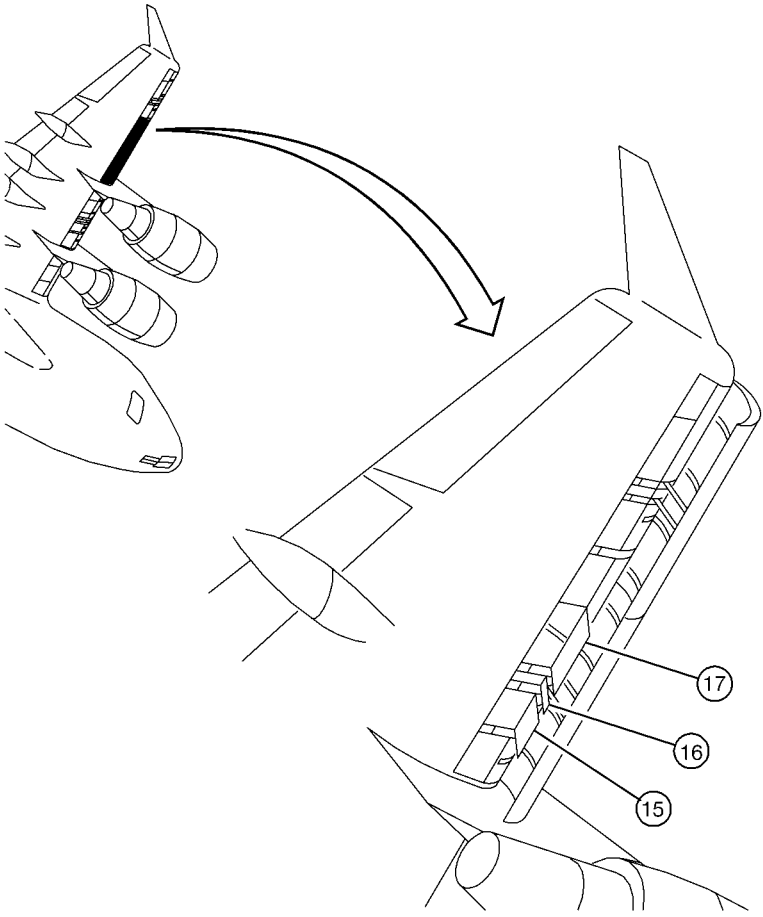
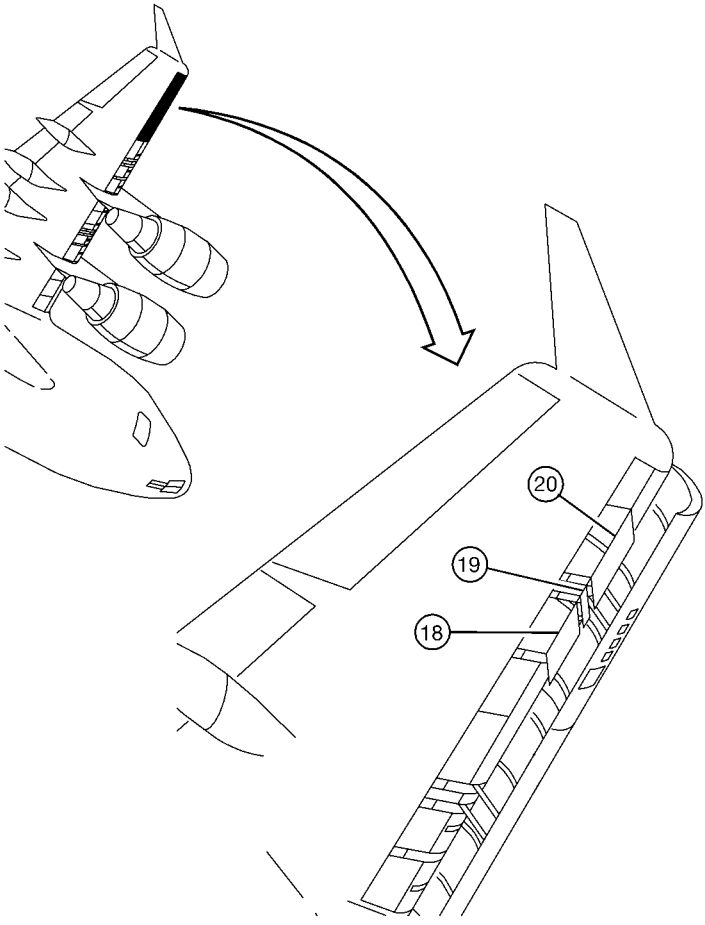


26

INDEX NO.	DOOR NO.	FASTENER QTY	LATCH QTY
15	633EBD	4	4
16	633FBD	2	2
17	633GBD	5	5

ICN-88277-G2780064-002-01



26

INDEX NO.	DOOR NO.	FASTENER QTY	LATCH QTY
18	633JBD	-	4
19	633KBD	-	2
20	633LBD	-	5

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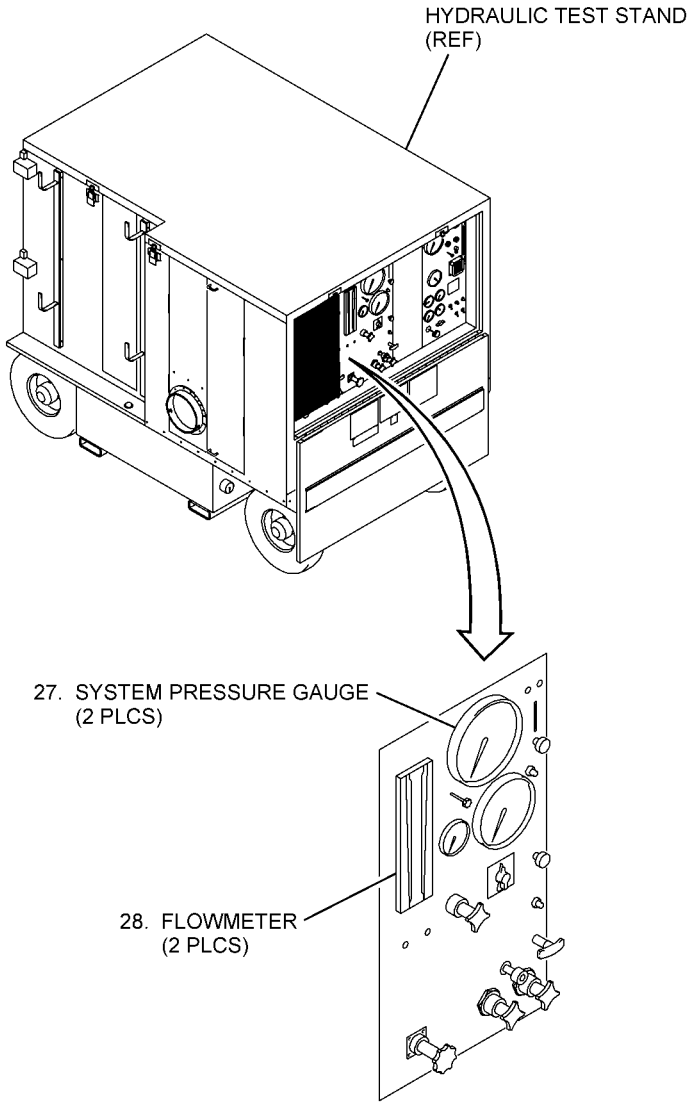
27. (D) Slowly increase **SYSTEM PRESSURE** gauges on hydraulic test stand for systems two and three to **2000 PSI**.

28. (D) Slowly increase **FLOWMETER** volume on hydraulic test stand for systems two and three to **30 GPM**.



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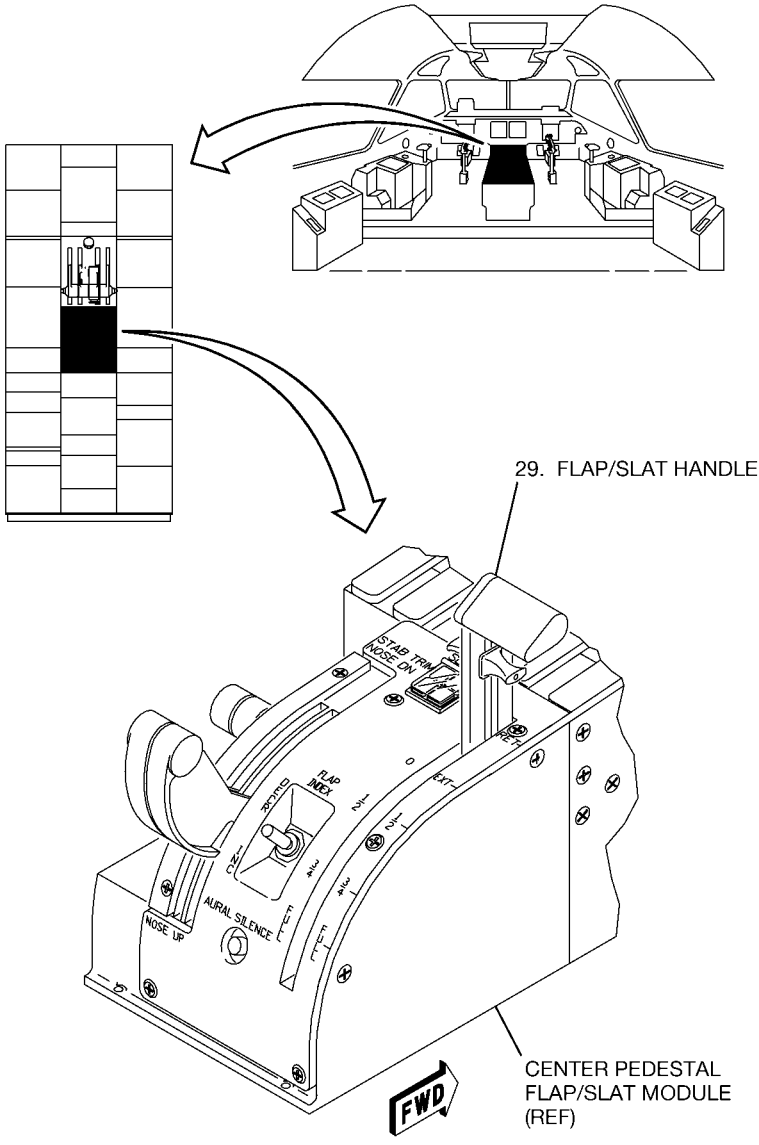
29. (C) Set flap/slat handle on center pedestal flap/slat module to **0/EXT** position.



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03-2. ALTERNATE SLAT ACTUATION BLEED.

WARNING

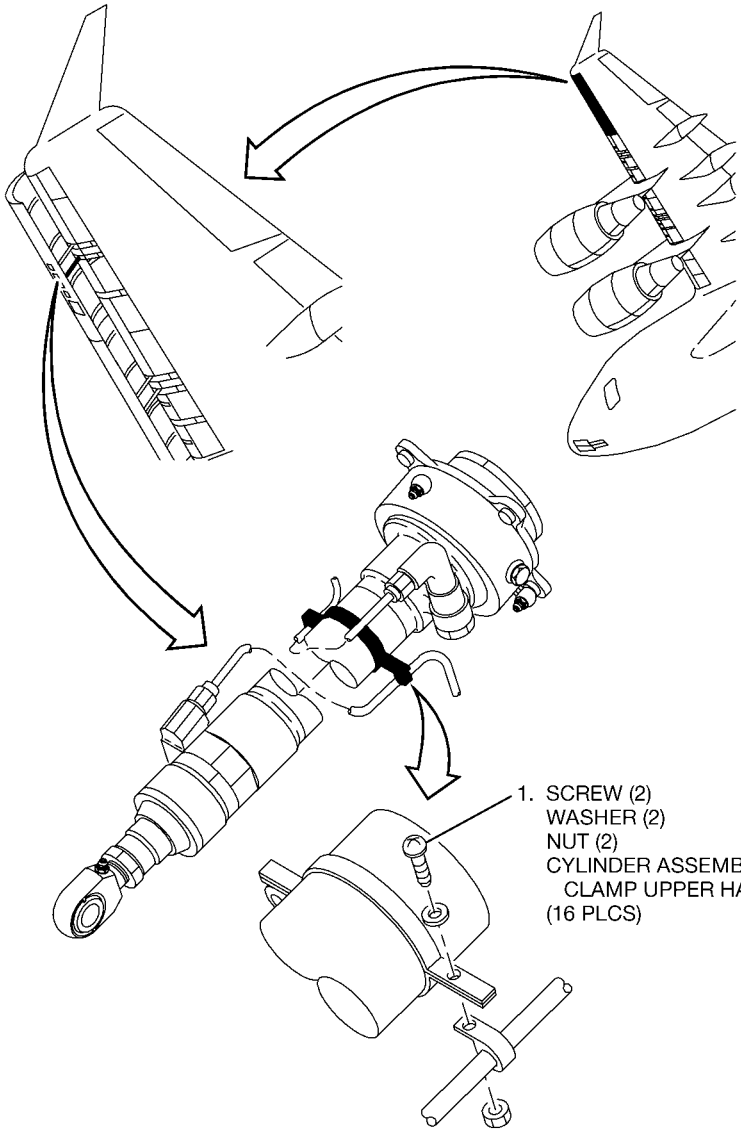
Pressures being applied are 2000 psi for pressure with 30 gallons per minute, and 42 psi for return pressure. Extreme caution must be taken when disconnecting hydraulic lines under pressure in order to prevent serious injury to personnel and damage to equipment.

NOTE

- To completely bleed air from slat actuation system, each linear actuating cylinder assembly hydraulic line must be disconnected and allowed to drain approximately 1 to 1.5 gallons of hydraulic fluid or until air bubbles are no longer visible and fluid drains clear.

- Hydraulic system 2 operates inboard linear actuating cylinder assembly of each slat assembly. Hydraulic system 3 operates outboard linear actuating cylinder assembly of each slat assembly.

1. (A,B) Remove screws, washers, nuts, and cylinder assembly clamp upper half.



(TYPICAL)

ICN-88277-G2780069-002-01

WARNING

Inadvertently disconnecting hydraulic pressure lines will cause serious personal injury and damage to aircraft.

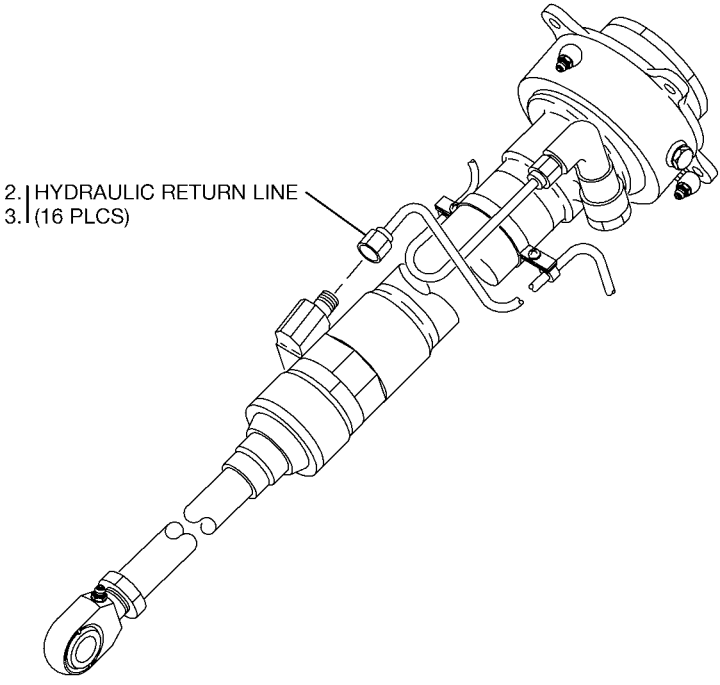
NOTE

Sequence for bleeding each cylinder assembly shall be accomplished beginning with furthest outboard cylinder assembly.

2. (A,B) Disconnect hydraulic return lines.

   - Verify no air bubbles are present in hydraulic fluid.

3. (A,B) Connect hydraulic return lines and torque **135-145 in-lb**.



(TYPICAL)

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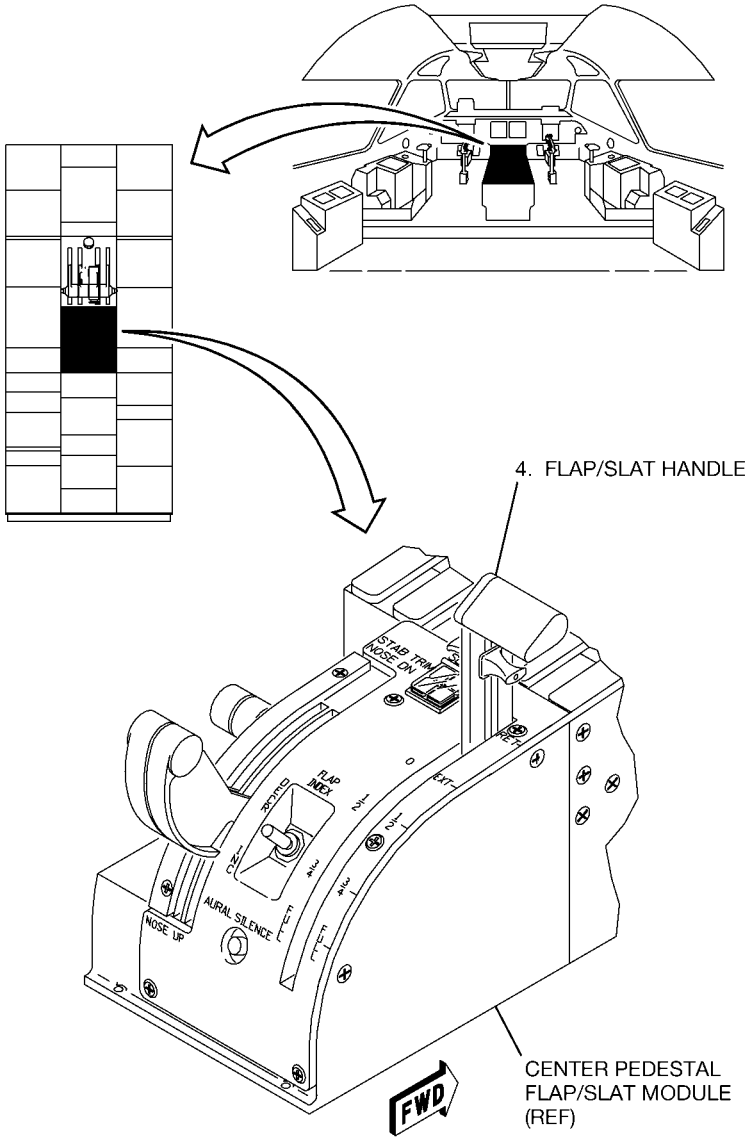
4. (C) Set flap/slat handle on center pedestal flap/slat module to **UP/RET** position.



ICN-88277-G2780071-002-01

WARNING

Inadvertently disconnecting hydraulic pressure lines will cause serious personal injury and damage to aircraft.

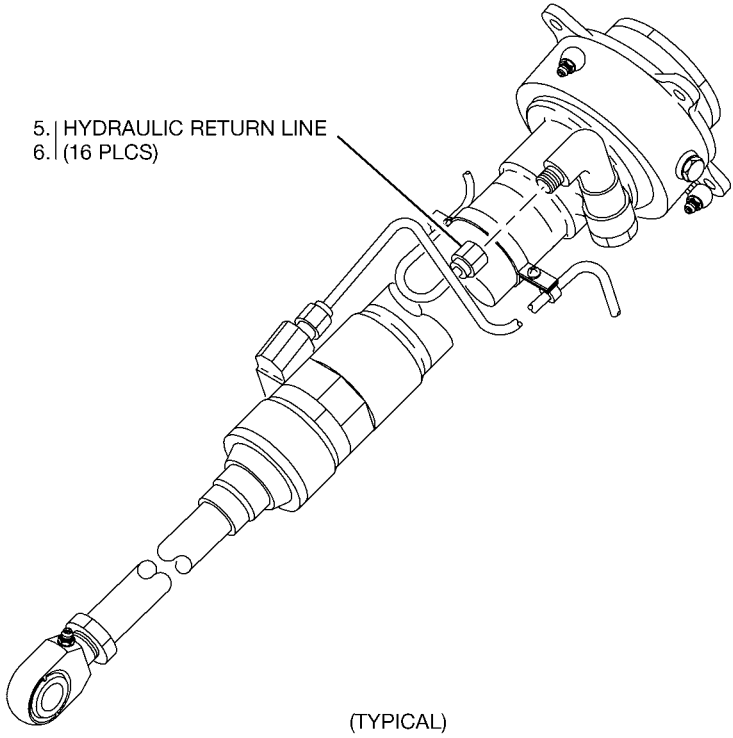
NOTE

Sequence for bleeding each cylinder assembly shall be accomplished beginning with furthest outboard cylinder assembly.

5. (A,B) Disconnect hydraulic return lines.

   - Verify no air bubbles are present in hydraulic fluid.

6. (A,B) Connect hydraulic return lines and torque **135-145 in-lb**.



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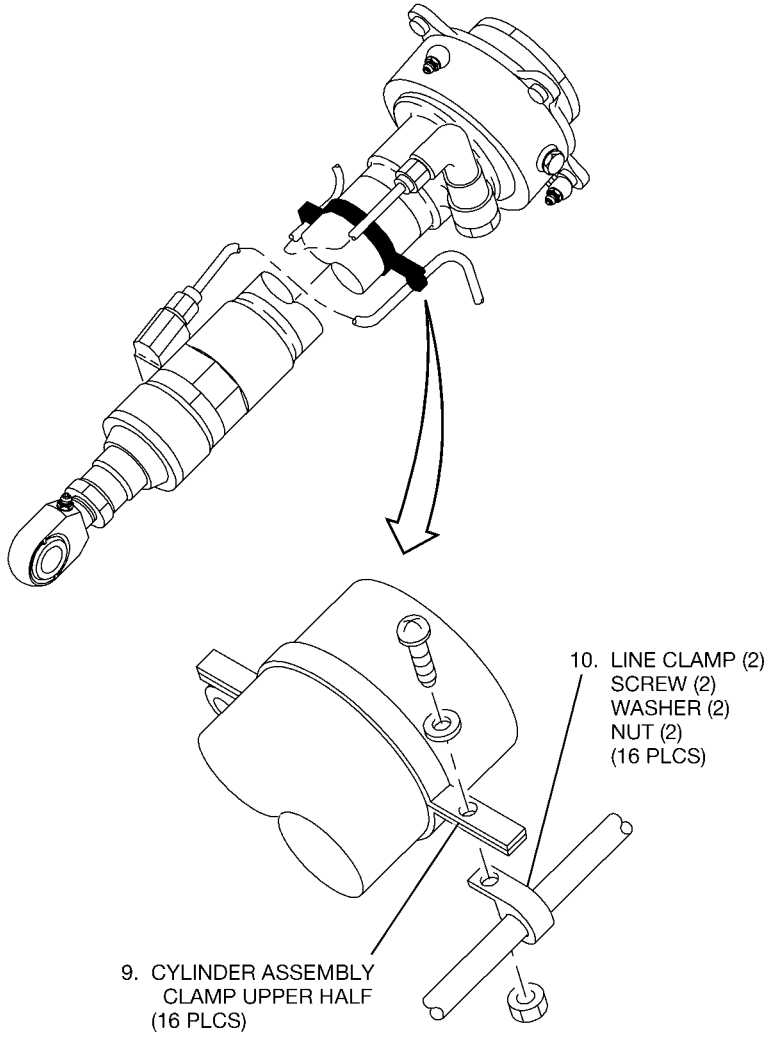
7. Perform faying surface sealing using MIL-PRF-81733 (TO 1300i-23, Chapter 1, Section III).

8. Perform wet fastener installation using AMS 3265 (TO 1300i-23, Chapter 1, Section III).

9. (A,B) Position cylinder assembly clamp upper half.

10. (A,B) Position line clamps under cylinder assembly clamp and install screws, washers, and nuts.



(TYPICAL)

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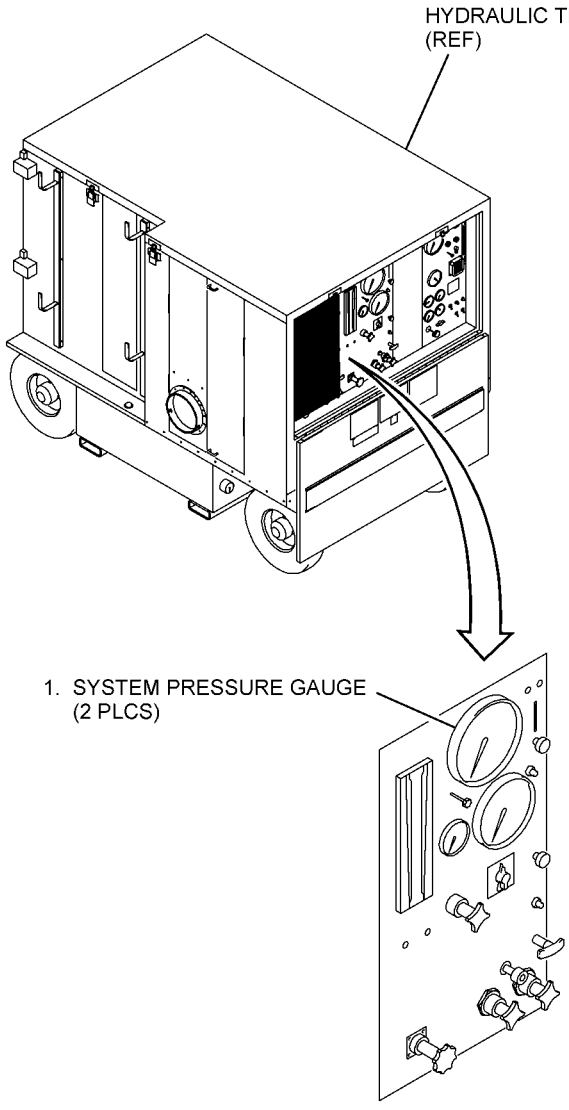
03-3. FOLLOW-ON MAINTENANCE.

1. (D) Decrease **SYSTEM PRESSURE** gauges on hydraulic test stand for systems two and three to **0 PSI**.



ICN-88277-G2780074-003-01

CAUTION

When winds are above 14 knots, aircraft ground safety locks shall be installed per 10-10-00, para 1-12 prior to bleeding down the gust damper system(s). Failure to comply may cause damage to control surfaces.

2. (C) Open access door (265TZM(AT)).

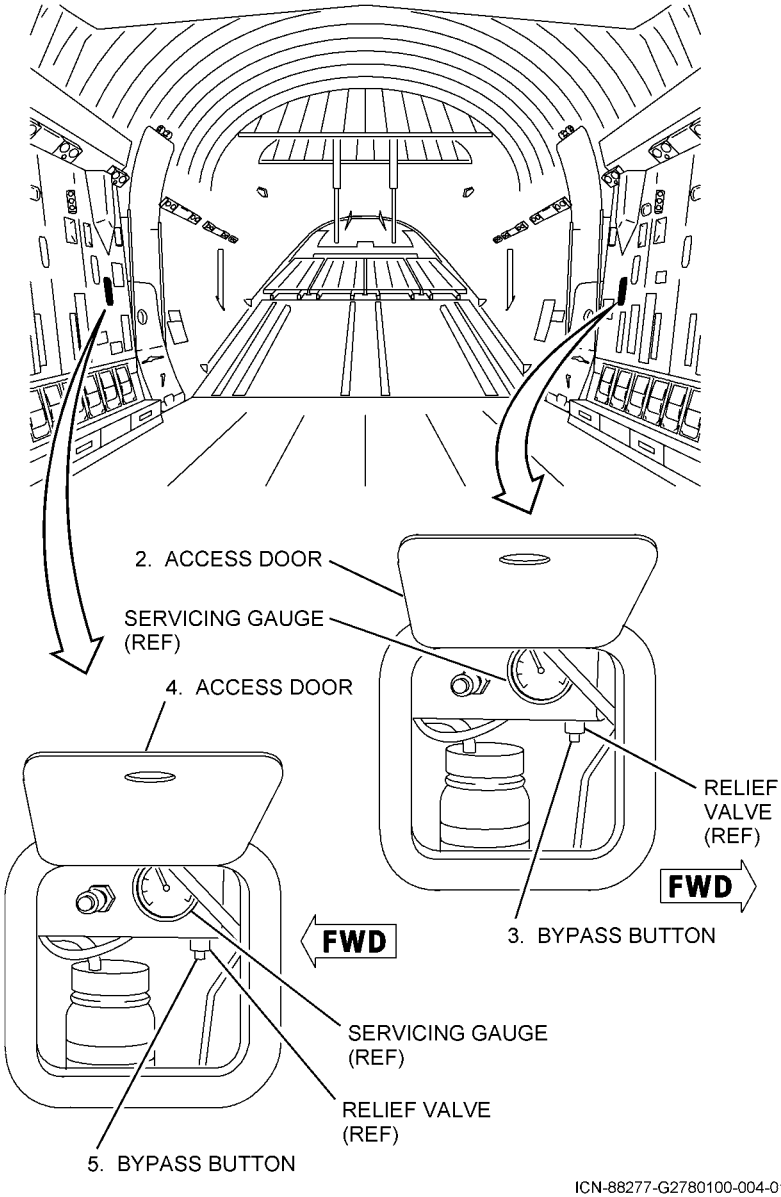
3. (C) Press bypass button on hydraulic system 2 relief valve until servicing gauge stops decreasing.

   - Servicing gauge reads nitrogen precharge (TO 1300i-2-12JG-29-1, 12-29-00, para 1-21).

4. (C) Open access door (266TZM(AT)).

5. (C) Press bypass button on hydraulic system 3 relief valve until servicing gauge stops decreasing.

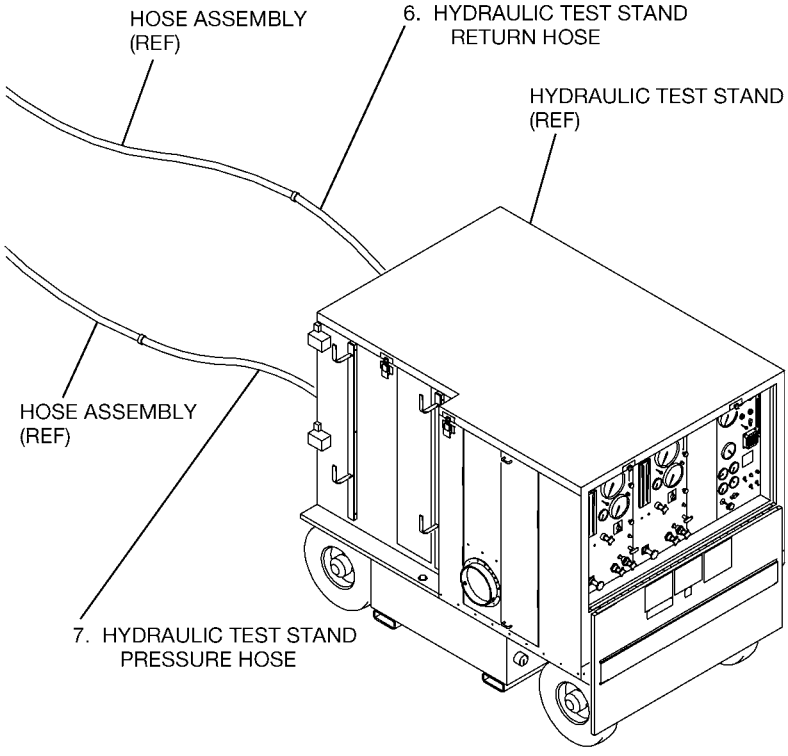
   - Servicing gauge reads nitrogen precharge (TO 1300i-2-12JG-29-1, 12-29-00, para 1-21).



ICN-88277-G2780100-004-01

6. (A,B) Disconnect hydraulic test stand return hose from hose assembly (PN 17G010064-1).

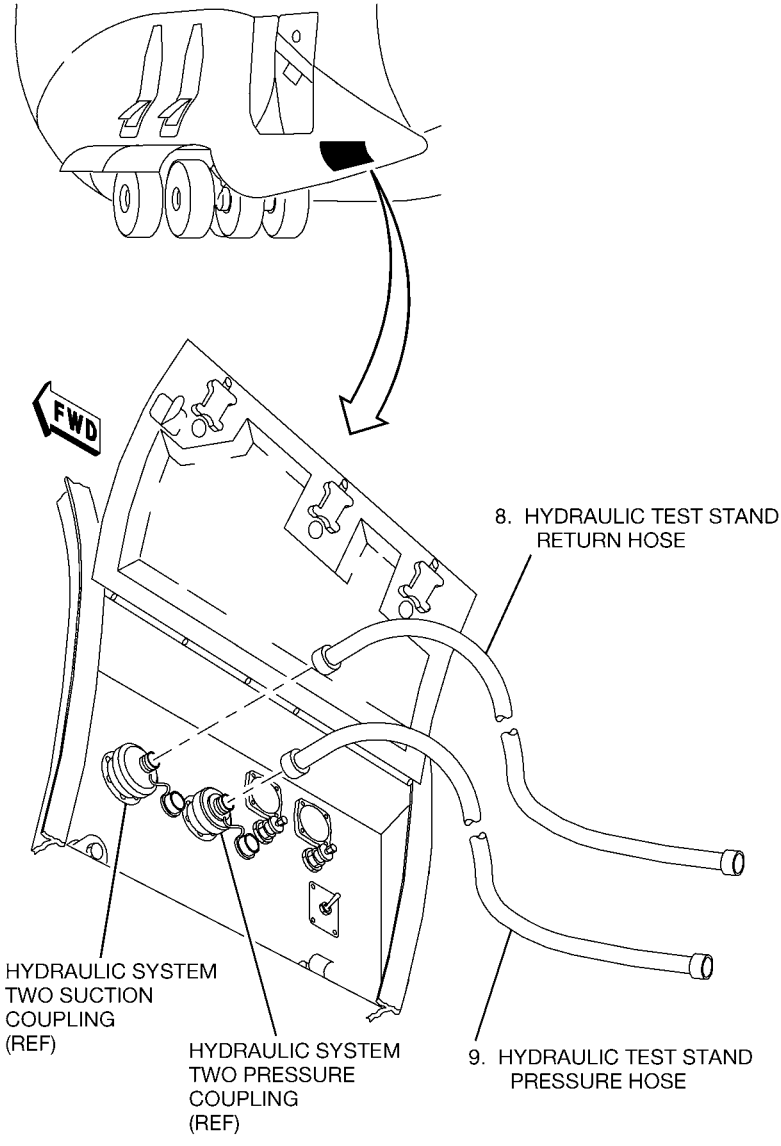
7. (A,B) Disconnect hydraulic test stand pressure hose from hose assembly (PN 17G010063-1).



ICN-88277-G2780075-002-01

8. (A,B) Disconnect hydraulic test stand return hose from hydraulic system two suction coupling.

9. (A,B) Disconnect hydraulic test stand pressure hose from hydraulic system two pressure coupling.

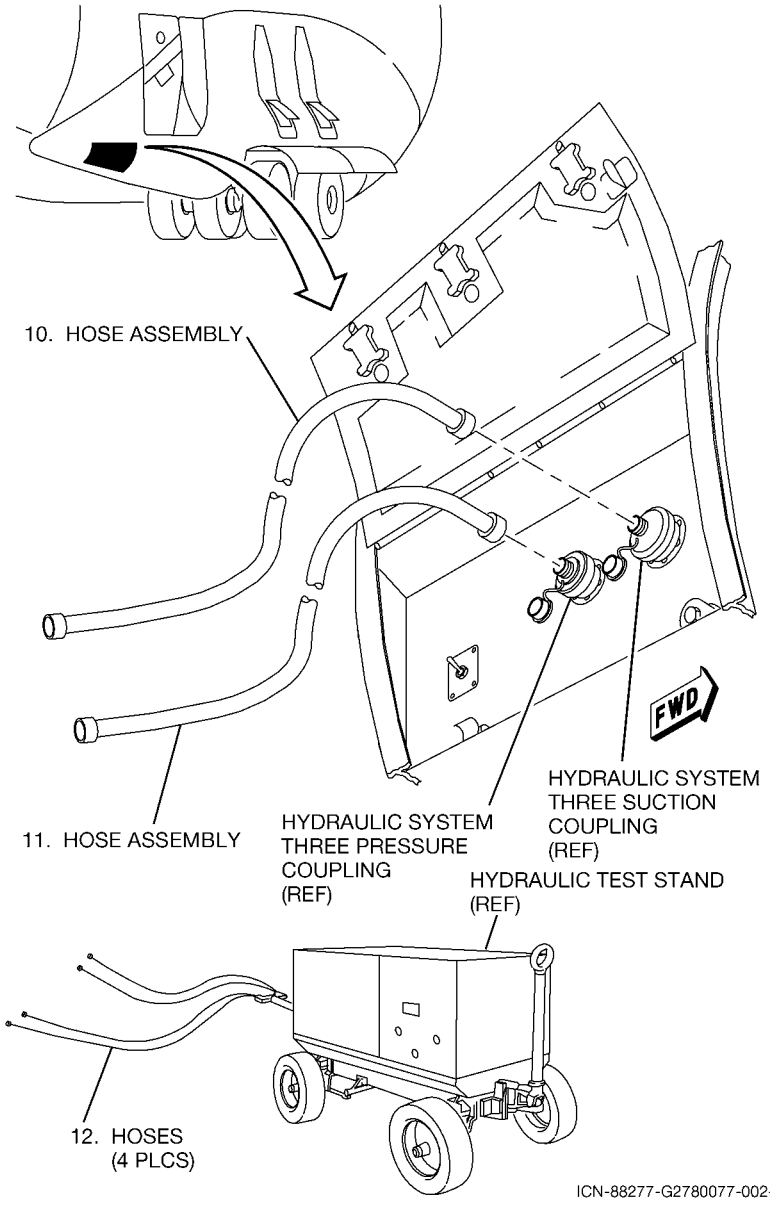


ICN-88277-G2780076-002-01

10. (A,B) Disconnect hose assembly (PN 17G010064-1) from hydraulic system three suction coupling.

11. (A,B) Disconnect hose assembly (PN 17G010063-1) from hydraulic system three pressure coupling.

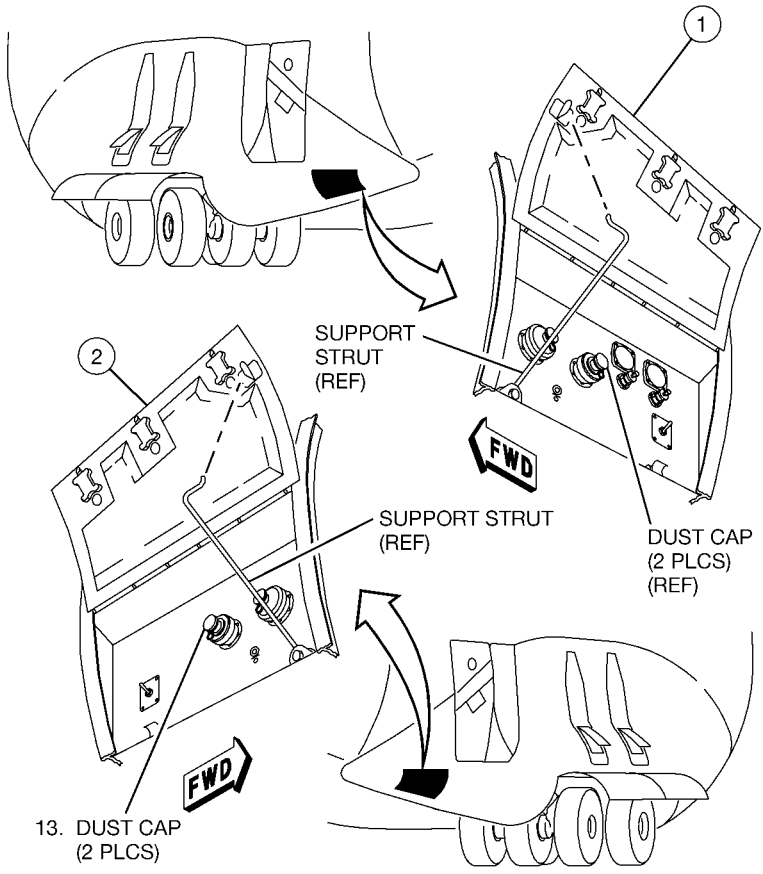
12. (A,B,D) Stow hydraulic test stand hoses.



ICN-88277-G2780077-002-01

13. (A,B) Install dust caps.

14. (A,B) Remove support strut; close and latch door.

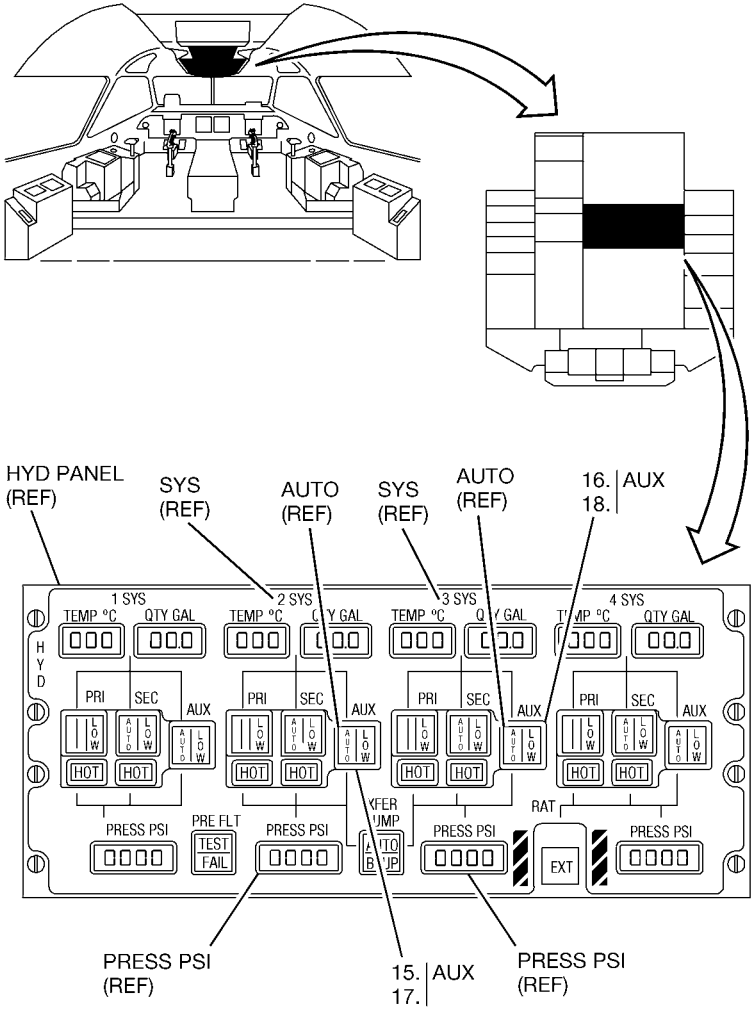


14.

INDEX NO.	HYDRAULIC SYSTEM	DOOR NO.	LATCH QTY
1	2	173CLD	3
2	3	183CRD	3

ICN-88277-G2780078-002-01

15. (C) Press **2 SYS, AUX** switchlight, on **HYD** panel.
    - **AUTO** light comes on.
    - **PRESS PSI** indicator reads 3800-4200.
16. (C) Press **3 SYS, AUX** switchlight.
    - **AUTO** light comes on.
    - **PRESS PSI** indicator reads 3800-4200.
17. (C) Press **2 SYS, AUX** switchlight.
    - **AUTO** light goes off.
18. (C) Press **3 SYS, AUX** switchlight.
    - **AUTO** light goes off.



ICN-88277-G2780101-002-01

CAUTION

When winds are above 14 knots, aircraft ground safety locks shall be installed per 10-10-00, para 1-12 prior to bleeding down the gust damper system(s). Failure to comply may cause damage to control surfaces.

19. (C) Open access door (265TZM(AT)).

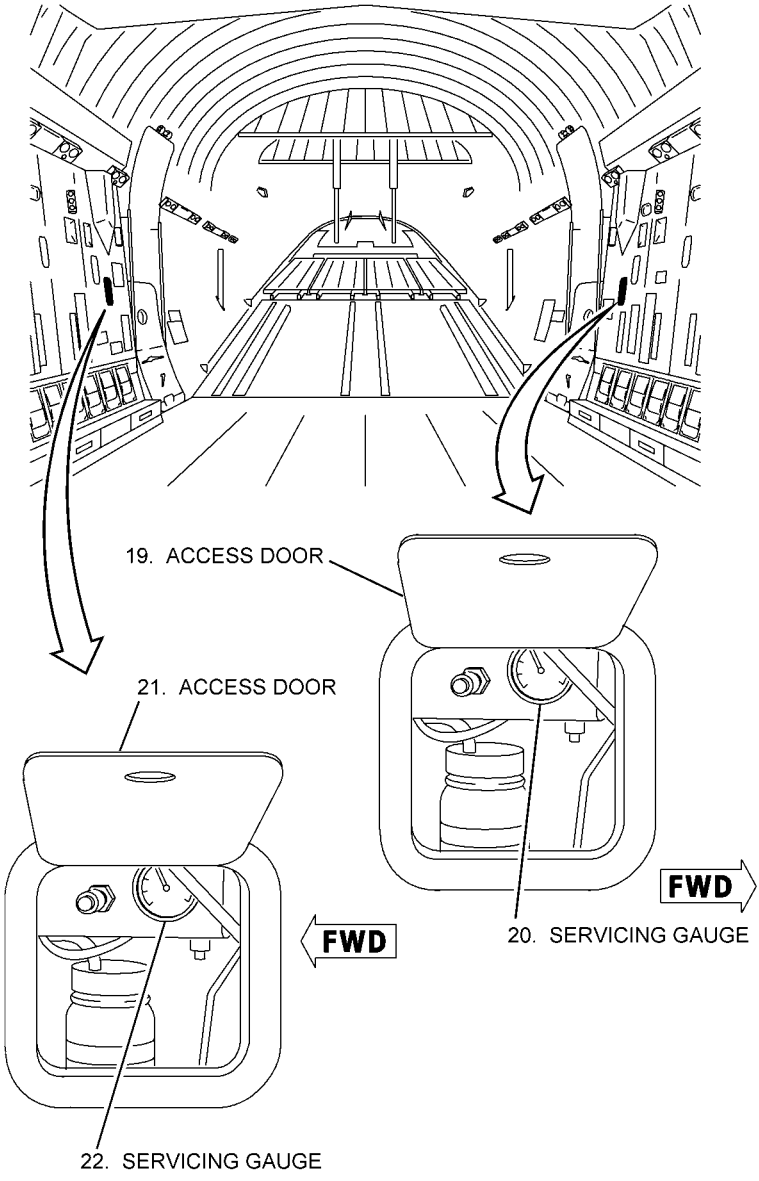
20. (C) Observe reading on hydraulic system 2 servicing gauge.

    - Servicing gauge reads 3800-4200 psi.

21. (C) Open access door (265TZM(AT)).

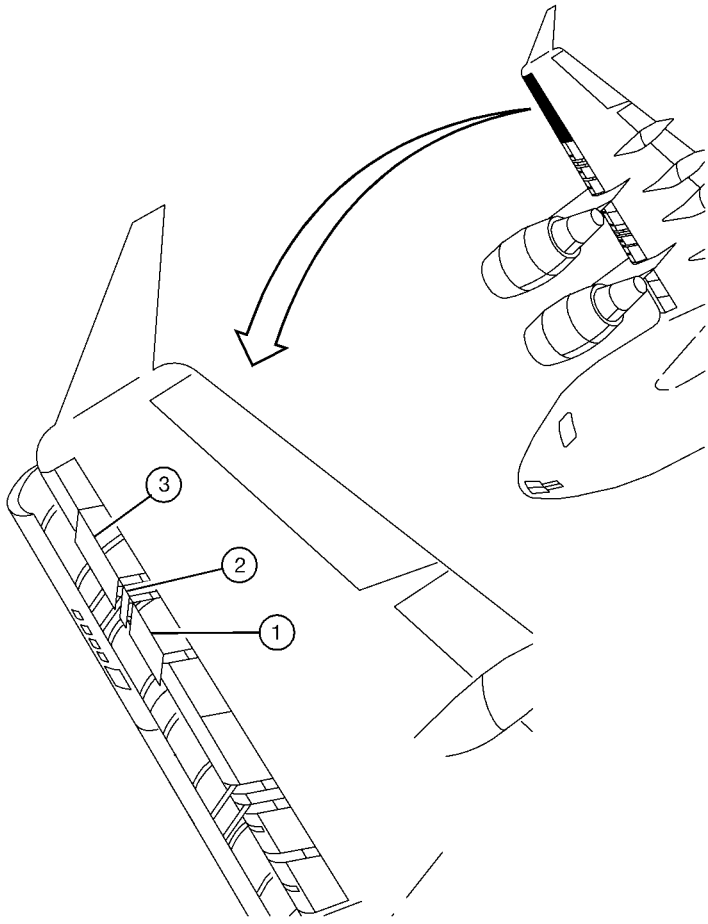
22. (C) Observe reading on hydraulic system 3 servicing gauge.

    - Servicing gauge reads 3800-4200 psi.



ICN-88277-G2780102-004-01

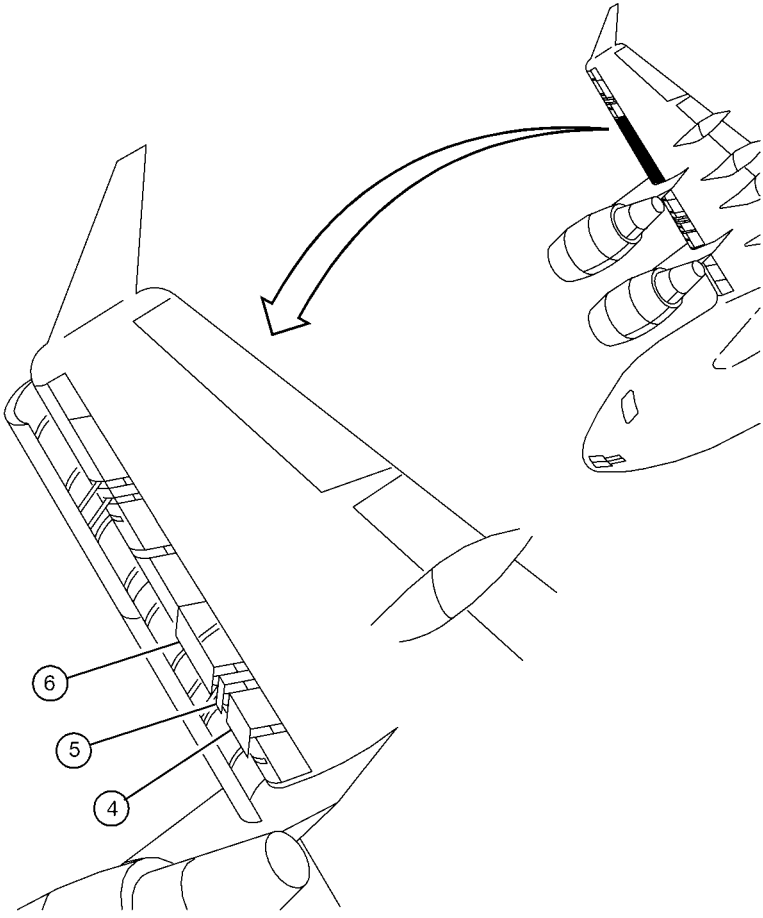
23. (A,B) Close and latch leading edge lower access doors; tighten fasteners.



23

INDEX NO.	DOOR NO.	FASTENER QTY	LATCH QTY
1	533JBD	-	4
2	533KBD	-	2
3	533LBD	-	5

ICN-88277-G2780079-002-01



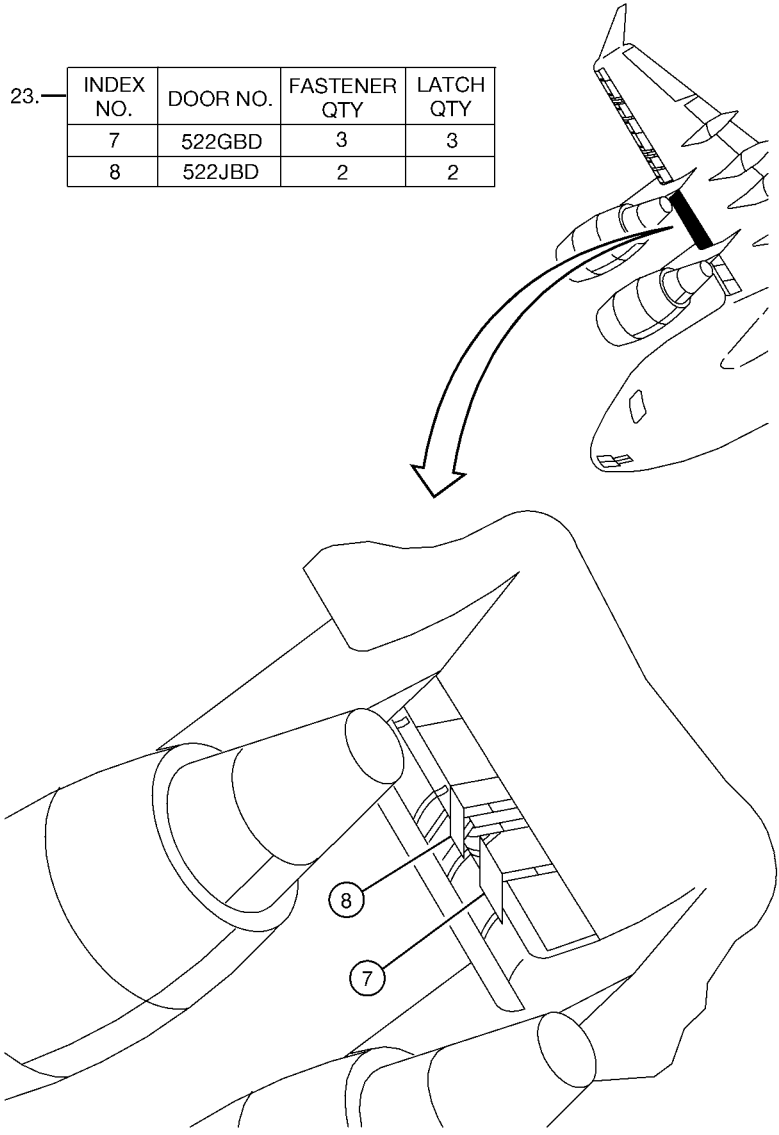
23

INDEX NO.	DOOR NO.	FASTENER QTY	LATCH QTY
4	533EBD	4	4
5	533FBD	2	2
6	533GBD	-	5

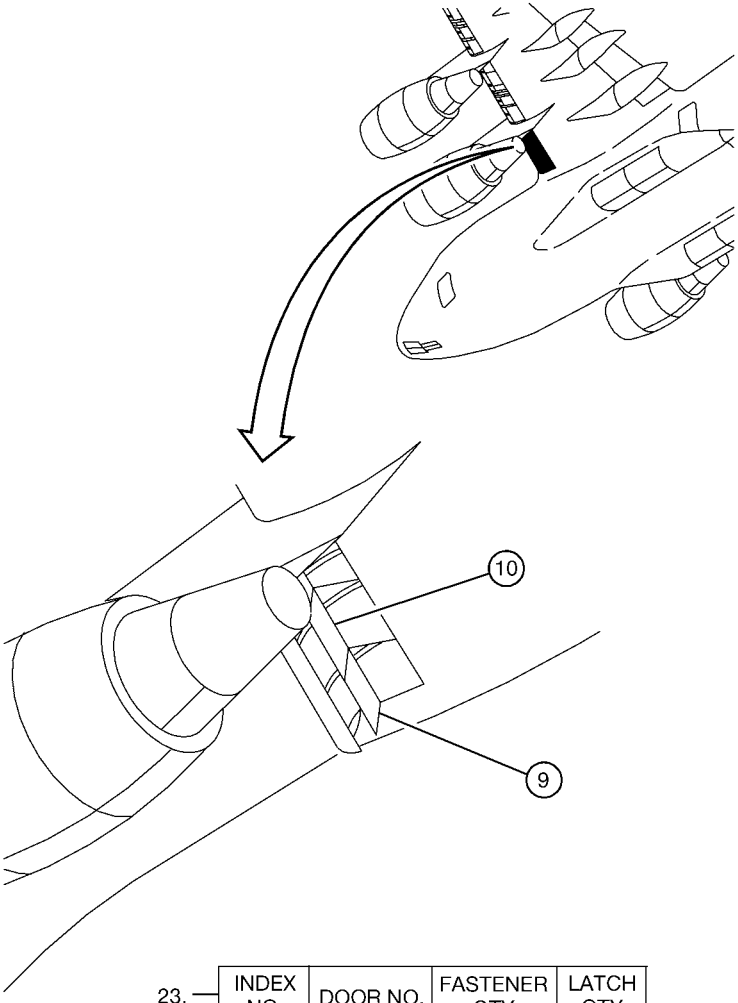
ICN-88277-G2780080-002-01

23.

INDEX NO.	DOOR NO.	FASTENER QTY	LATCH QTY
7	522GBD	3	3
8	522JBD	2	2



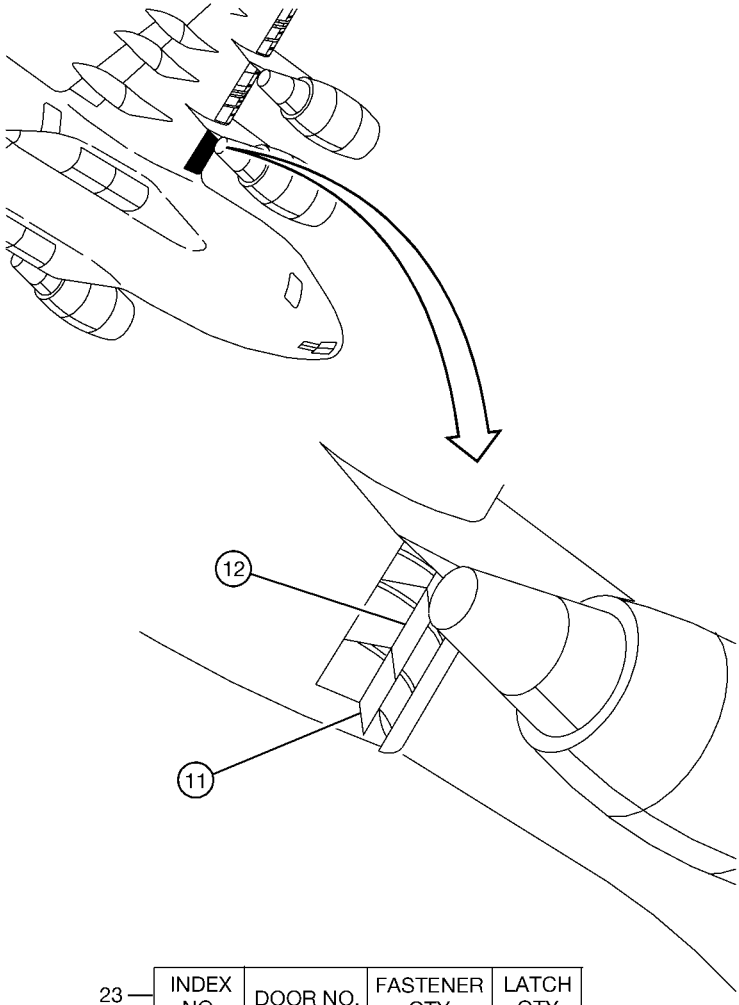
ICN-88277-G2780081-002-01



23.

INDEX NO.	DOOR NO.	FASTENER QTY	LATCH QTY
9	512ABD	5	5
10	512BBD	5	5

ICN-88277-G2780082-002-01



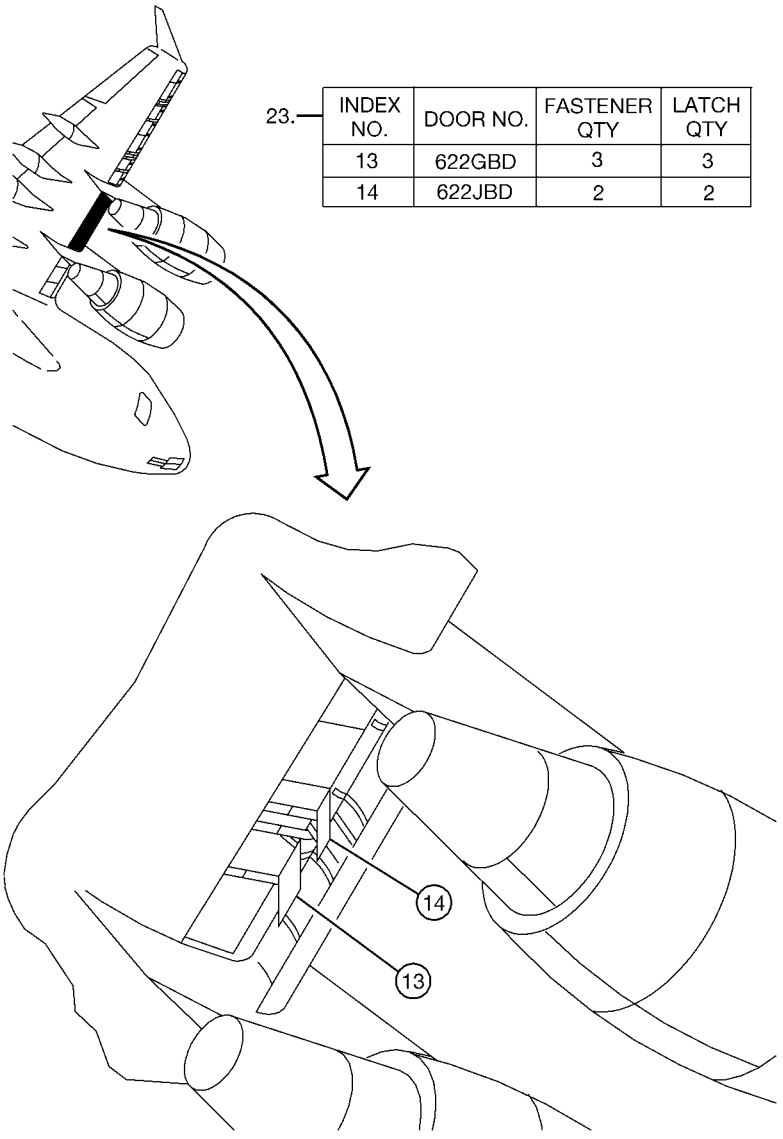
23

INDEX NO.	DOOR NO.	FASTENER QTY	LATCH QTY
11	612ABD	5	5
12	612BBD	5	5

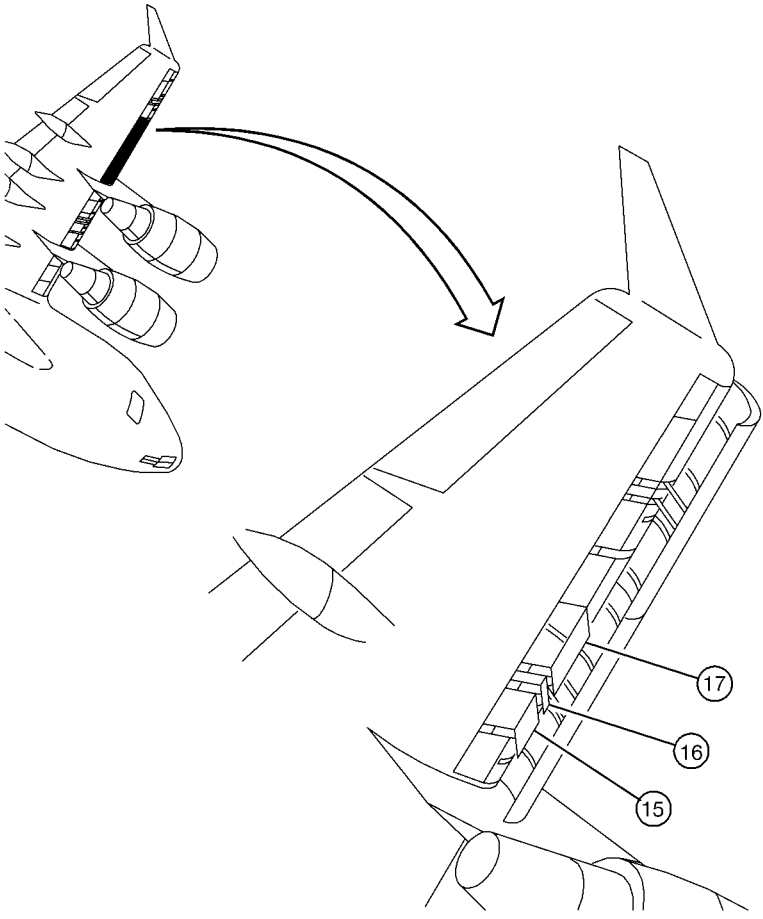
ICN-88277-G2780083-002-01

23.

INDEX NO.	DOOR NO.	FASTENER QTY	LATCH QTY
13	622GBD	3	3
14	622JBD	2	2



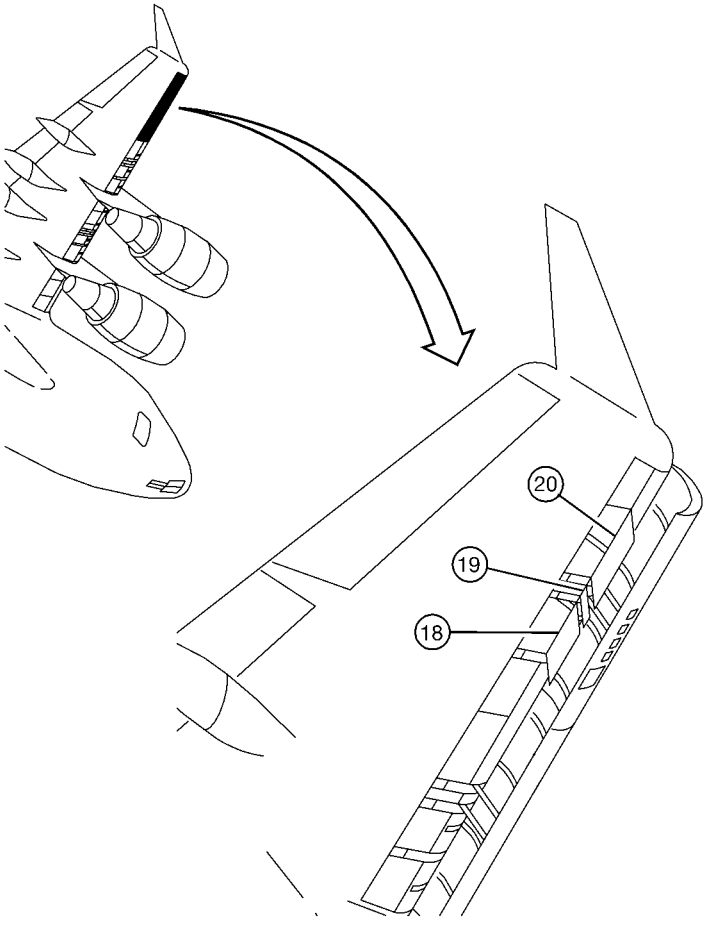
ICN-88277-G2780084-002-01



23

INDEX NO.	DOOR NO.	FASTENER QTY	LATCH QTY
15	633EBD	4	4
16	633FBD	2	2
17	633GBD	5	5

ICN-88277-G2780085-002-01



23

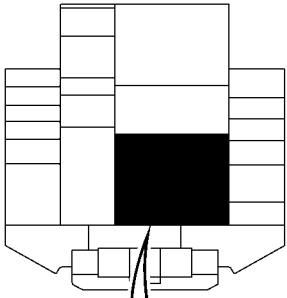
INDEX NO.	DOOR NO.	FASTENER QTY	LATCH QTY
18	633JBD	-	4
19	633KBD	-	2
20	633LBD	-	5

ICN-88277-G2780086-002-01

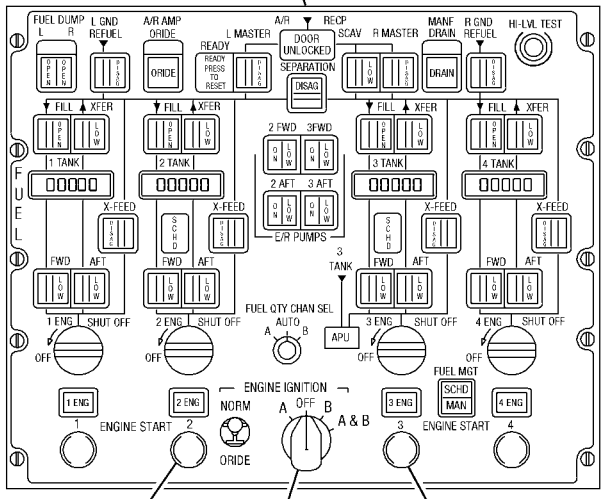
24. (C) Remove warning tag from **ENGINE START 2** switch on **FUEL** panel.

25. (C) Remove warning tag from **ENGINE START 3** switch.

26. (C) Remove warning tag from **ENGINE IGNITION** switch.



FUEL PANEL
(REF)



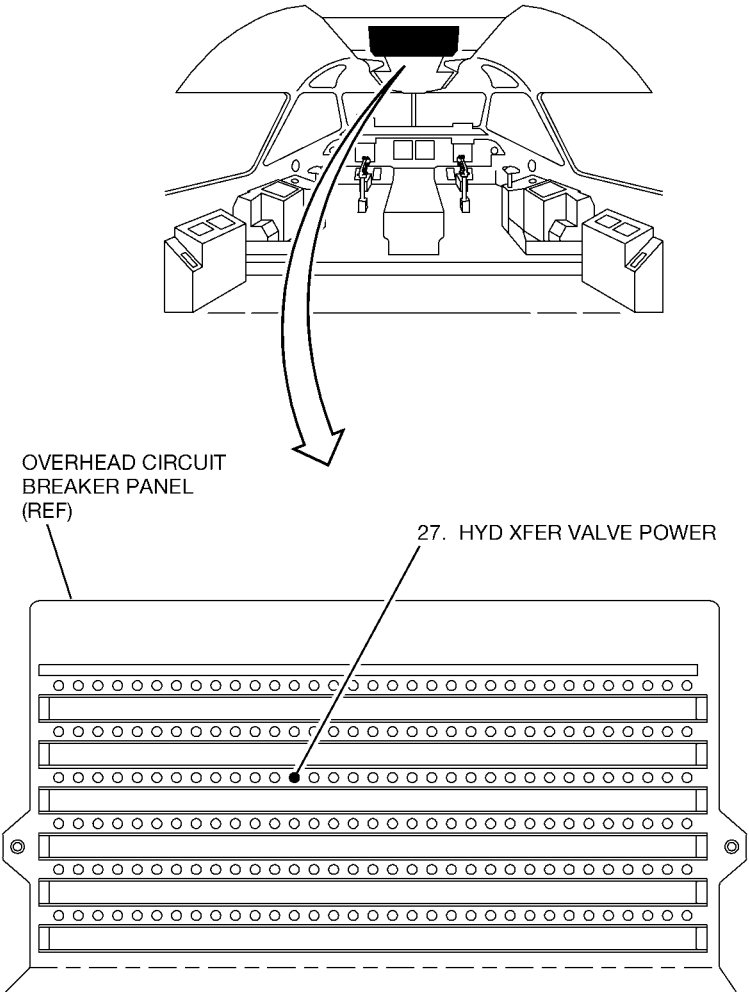
24. ENGINE START 2

25. ENGINE START 3

26. ENGINE IGNITION

ICN-88277-G2780087-003-01

27. (C) Remove warning tag and close **HYD XFER VALVE POWER** circuit breaker on overhead circuit breaker panel, row **C**, column **13**.

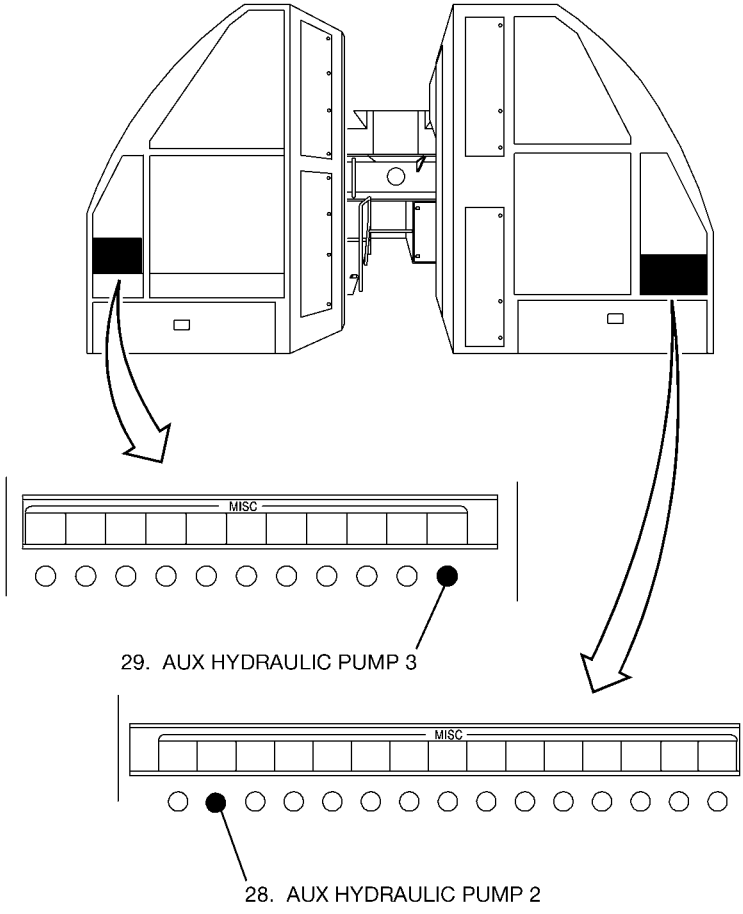


ICN-88277-G2780088-002-01

28. (C) Remove warning tag and close **AUX HYDRAULIC PUMP 2** circuit breaker on Electrical Power Center (EPC), row **LL**, column **69**.

29. (C) Remove warning tag and close **AUX HYDRAULIC PUMP 3** circuit breaker on EPC, row **LL**, column **11**.

30. Perform maintenance interphone shutdown (23-41-02, task 02-4).



ICN-88277-G2780089-002-01

CONTROLS AND SURFACES SYSTEM REPAIR (27-81-01)

GENERAL MAINTENANCE INPUT CONDITIONS:

Applicability:

	Task
All	All

Additional information:

This procedure consists of the following tasks:

01-1. Repair controls and surfaces system by replacing slat tandem directional control valve actuator input connecting link.
01-2. Repair controls and surfaces system by replacing stop pin.
01-3. Repair controls and surfaces system by replacing down stop.
01-4. Repair controls and surfaces system by replacing up stop.

Additional data:

	Task
TO 1300i-23	01-2, 01-3, 01-4
TO 1300i-2-27JG-50-1	01-2, 01-4
TO 1300i-2-27JG-80-1	01-1
TO 1300i-2-27JG-80-3	01-4
TO 1300i-2-27JG-80-6	01-1

Personnel recommended:

	Task
One	All

Safety conditions:

	Task
NA	--

Support equipment:

Nomenclature	PN	Specification	Qty	Task
Lock, Aircraft Ground Safety, Flap Control Handle	17G140025-1	--	1	01-2, 01-4
Wrench, Torque	--	(0-150 in-lb)	1	01-2

Supplies:

Nomenclature	PN	Specification	Qty	Task
Sealant	PR-1775 B-2	AMS 3265	AR	01-2, 01-3, 01-4
Sealant	PR-1775 B-2	MIL-PRF-81733	AR	01-3, 01-4
Pin, Cotter	MS24665-151	--	2	01-1

Nomenclature	PN	Specification	Qty	Task
Tag, Warning	--	--	2	01-1, 01-2, 01-4

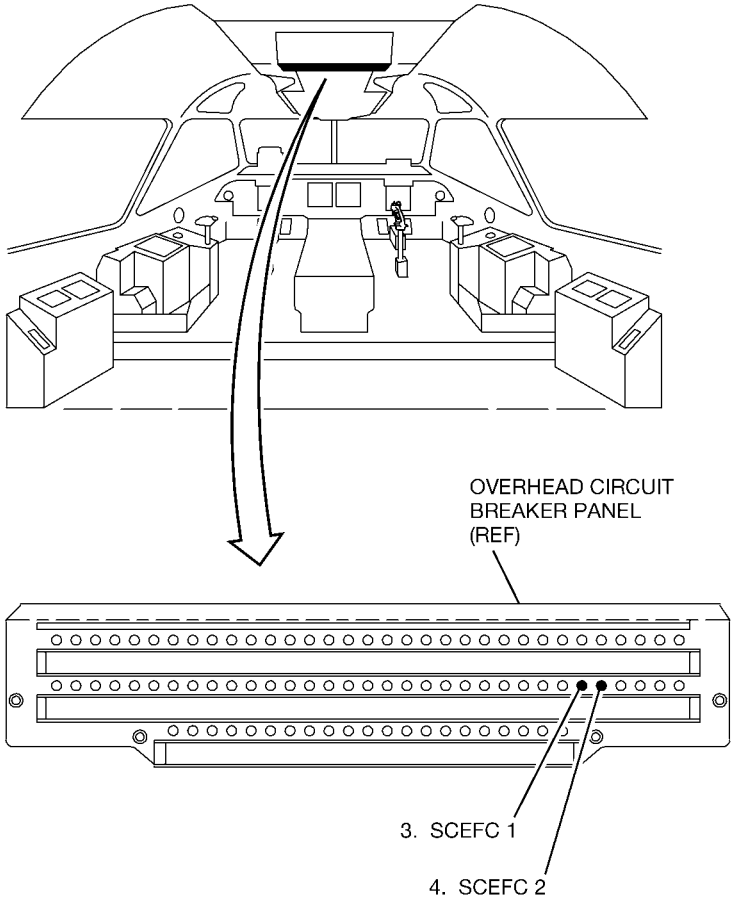
01-1. REPAIR CONTROLS AND SURFACES SYSTEM BY REPLACING SLAT TANDEM DIRECTIONAL CONTROL VALVE ACTUATOR INPUT CONNECTING LINK.

1. Review ″Section 1 (General Information)″ of this TO for system general warnings, cautions, and notes.

2. Review task ″General Maintenance Input Conditions″ page for task specific safety conditions.

3. Open **SCEFC 1** circuit breaker on overhead circuit breaker panel, row **H**, column **28**, and attach warning tag.

4. Open **SCEFC 2** circuit breaker on overhead circuit breaker panel, row **H**, column **29**, and attach warning tag.



ICN-88277-G2781001-004-01

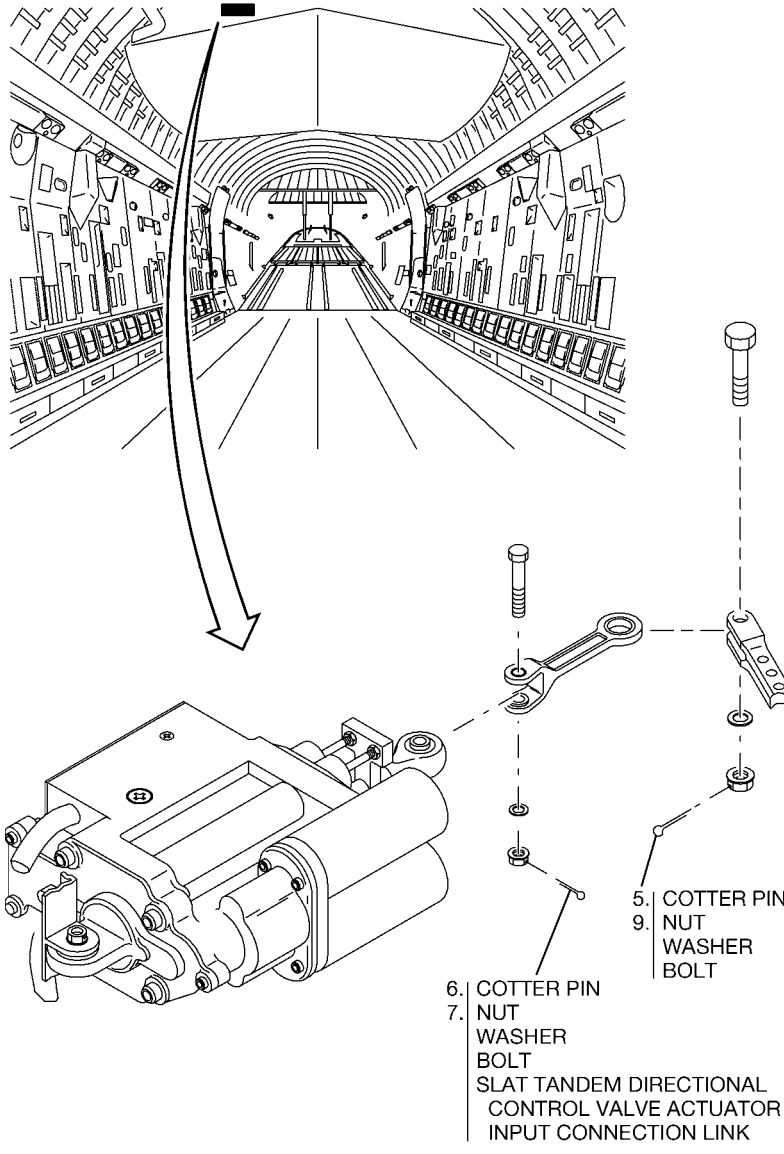
5. Remove cotter pin, nut, washer, and bolt.

6. Remove cotter pin, nut, washer, bolt, and slat tandem directional control valve actuator input connecting link.

7. Position link; install bolt, washer, nut, and cotter pin.

8. Perform slat tandem control valve actuator adjustment (27-82-13).

9. Install bolt, washer, nut, and cotter pin.



ICN-88277-G2781002-002-01

10. Remove warning tag and close **SCEFC 1** circuit breaker on overhead circuit breaker panel, row **H**, column **28**.

11. Remove warning tag and close **SCEFC 2** circuit breaker on overhead circuit breaker panel, row **H**, column **29**.

12. Perform slats system operational checkout (27-80-01).