

ENAE464 - 0202

Lab02: Pressure Drag and Lift on an Airfoil Model

Due on February 27th, 2026 at 11:59 PM

Dr. Silbaugh, 02:00 PM

Mikołaj Kostrzewa & Vai Srivastava

Experiment Performed: February 20th, 2026
Report Submitted: February 27th, 2026

February 27th, 2026

Enclosed is the technical report for the *Pressure Drag and Lift on an Airfoil Model* laboratory experiment. This report presents the experimental methodology, results, and analysis of aerodynamic characteristics observed on the surface of a scale model of a NACA airfoil, including lift coefficient and drag coefficient distributions, aerodynamic force analyses, and comparison with published NACA data. Please feel free to contact us with any questions regarding the contents of this report.

Respectfully,
Mikołaj Kostrzewa & Vai Srivastava

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1 Abstract

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2 Introduction

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3 Experimental Apparatus and Procedures

3.1 Experimental Apparatus

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3.2 Operating Technique

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4 Results

4.1 Operating Conditions

Table 4.1.1: Ambient Room Conditions during Experiment

Quantity	Value	Uncertainty
Room Temperature, T	25 °C	±0.5 °C
Room Pressure, P	1009.0 hPa	±0.05 hPa

The density of air (ρ) can be calculated using the Ideal Gas Law:

$$\rho = \frac{P}{RT} \quad (4.1.1)$$

where:

- P is the absolute pressure (in SI units: Pascals Pa)
- T is the absolute temperature (in Kelvin K)
- R is the specific gas constant for dry air, $R = 287.05 \text{ J kg}^{-1} \text{ K}^{-1}$

Given measurements:

$$P = 1009.0 \text{ hPa} = 100900 \text{ Pa} \quad (4.1.2)$$

$$T = 25 \text{ C} = 25 + 273.15 = 298.15 \text{ K} \quad (4.1.3)$$

$$R = 287.05 \frac{\text{J}}{\text{kg K}} \quad (4.1.4)$$

$$\rho = \frac{100\,900 \text{ Pa}}{287.05 \text{ J kg}^{-1} \text{ K}^{-1}} \times 298.15 \text{ K} \quad (4.1.5)$$

Calculate:

$$\rho = \frac{100900}{287.05 \times 298.15} \quad (4.1.6)$$

$$= \frac{100900}{85542.5} \quad (4.1.7)$$

$$\approx 1.18 \frac{\text{kg}}{\text{m}^3} \quad (4.1.8)$$

Thus, the ambient air density during the experiment is:

$\rho = 1.18 \frac{\text{kg}}{\text{m}^3}$

(4.1.9)

4.2 Measurements

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4.3 Quantities of Interest

4.3.1 Incoming Wind Tunnel Airflow

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4.3.2 Coefficient of Pressure

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4.3.3 Pressure Drag and Lift Force

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4.3.4 Coefficient of Lift

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4.3.5 Coefficient of Drag

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5 Analysis and Discussion

5.1 Experimental Aerodynamic Characteristics vs. Published NACA Data

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5.2 Experimental Lift Curve vs. Published NACA Data

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5.3 Experimental Drag Polar vs. Published NACA Data

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6 Summary and Conclusions

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7 References

- [1] M. S. Selig, *UIUC airfoil data site*, University of Illinois at Urbana-Champaign: Department of Aeronautical and Astronautical Engineering. [Online]. Available: https://m-selig.ae.illinois.edu/ads/coord_database.html (see p. 8).
- [2] M. Drela and H. Youngren, *XFOIL, Subsonic airfoil development system*, Massachusetts Institute of Technology: Department of Aeronautics and Astronautics. [Online]. Available: <https://web.mit.edu/drela/Public/web/xfoil> (see p. 9).
- [3] J. D. Anderson and C. P. Cadou, *Fundamentals of Aerodynamics*, 7th ed. McGraw-Hill Education, March 17, 2023, ISBN: 9781266076442.

8 Appendices

8.1 Data

Table 8.1.1: NACA 4412 Airfoil Coordinates and the Corresponding Tap Locations on the Model Airfoil [1]

<code>./data/naca_4412_airfoil_coords_and_taps.csv</code>			
x/c	y/c	Tap #	DSA Channel #
1	0	—	01
0.95	0.0147	—	—
0.90	0.0271	—	—
0.80	0.0489	01	02
0.70	0.0669	—	—
0.60	0.0814	02	03
0.50	0.0919	—	—
0.40	0.0980	03	04
0.30	0.0976	04	05
0.25	0.0941	—	—
0.20	0.0880	05	06
0.15	0.0789	—	—
0.10	0.0659	06	07
0.075	0.0576	—	—
0.05	0.0473	07	08
0.025	0.0339	—	—
0.0125	0.0244	—	—
0	0	08	09
0.0125	-0.0143	—	—
0.025	-0.0195	—	—
0.05	-0.0249	09	10
0.075	-0.0274	—	—
0.10	-0.0286	—	—
0.15	-0.0288	10	11
0.20	-0.0274	—	—
0.25	-0.0250	11	12
0.30	-0.0226	—	—
0.35	-0.0203	12	13
0.40	-0.0180	—	—
0.50	-0.0140	13	14
0.60	-0.0100	—	—
0.65	-0.00825	14	15
0.70	-0.0065	—	—
0.80	-0.0039	—	—
0.90	-0.0022	—	—
0.95	-0.0016	—	—
1	0	—	16

Table 8.1.2: NACA 4412 Airfoil Drag Polar [2]

```
./data/xf-naca4412-il-1000000.csv
```

```

1 Xfoil polar. Reynolds number fixed. Mach number fixed
2 Polar key,xf-naca4412-il-1000000
3 Airfoil,naca4412-il
4 Reynolds number,1000000
5 Ncrit,9
6 Mach,0
7 Max C1/Cd,129.373
8 Max C1/Cd alpha,5.25
9 Url,http://www.airfoiltools.com/polar/csv?polar=xf-naca4412-il-1000000
```

α	C_l	C_d	C_{dp}	C_m	Top _{Xtr}	Bot _{Xtr}
-15.750	-0.8374	0.08373	0.08141	-0.0585	1.0000	0.0169
-15.500	-0.9127	0.06837	0.06591	-0.0687	1.0000	0.0166
-15.250	-1.0965	0.03328	0.03022	-0.0993	1.0000	0.0153
-15.000	-1.1161	0.03120	0.02803	-0.0956	1.0000	0.0154
-14.750	-1.1210	0.02977	0.02651	-0.0926	1.0000	0.0156
-14.500	-1.1215	0.02857	0.02523	-0.0896	1.0000	0.0159
-14.250	-1.1181	0.02751	0.02407	-0.0870	1.0000	0.0162
-14.000	-1.0990	0.02637	0.02282	-0.0871	0.9992	0.0166
-13.750	-1.0711	0.02533	0.02165	-0.0885	0.9979	0.0170
-13.500	-1.0462	0.02365	0.01985	-0.0903	0.9963	0.0177
-13.250	-1.0163	0.02288	0.01905	-0.0918	0.9951	0.0183
-13.000	-0.9847	0.02237	0.01850	-0.0933	0.9943	0.0189
-12.750	-0.9549	0.02183	0.01790	-0.0943	0.9930	0.0195
-12.500	-0.9260	0.02126	0.01724	-0.0952	0.9911	0.0201
-12.250	-0.8954	0.02078	0.01666	-0.0963	0.9894	0.0206
-12.000	-0.8682	0.01946	0.01525	-0.0976	0.9877	0.0214
-11.750	-0.8365	0.01894	0.01471	-0.0990	0.9866	0.0220
-11.500	-0.8038	0.01852	0.01424	-0.1004	0.9857	0.0227
-11.250	-0.7707	0.01808	0.01375	-0.1019	0.9849	0.0235
-11.000	-0.7369	0.01769	0.01328	-0.1035	0.9843	0.0242
-10.750	-0.7070	0.01745	0.01297	-0.1041	0.9819	0.0246
-10.500	-0.6803	0.01619	0.01161	-0.1049	0.9793	0.0257
-10.250	-0.6491	0.01569	0.01109	-0.1060	0.9775	0.0264
-10.000	-0.6172	0.01529	0.01065	-0.1071	0.9759	0.0271
-9.750	-0.5850	0.01491	0.01022	-0.1082	0.9742	0.0279
-9.500	-0.5547	0.01456	0.00981	-0.1089	0.9718	0.0287
-9.250	-0.5287	0.01426	0.00944	-0.1085	0.9665	0.0292
-9.000	-0.5023	0.01345	0.00855	-0.1085	0.9622	0.0299
-8.750	-0.4769	0.01285	0.00791	-0.1082	0.9574	0.0309
-8.500	-0.4513	0.01249	0.00752	-0.1078	0.9519	0.0317
-8.250	-0.4243	0.01214	0.00713	-0.1076	0.9474	0.0324

-8.000	-0.3979	0.01184	0.00678	-0.1073	0.9422	0.0333
-7.750	-0.3715	0.01155	0.00644	-0.1070	0.9363	0.0340
-7.500	-0.3442	0.01127	0.00609	-0.1068	0.9313	0.0345
-7.250	-0.3183	0.01080	0.00556	-0.1064	0.9249	0.0354
-7.000	-0.2921	0.01033	0.00505	-0.1061	0.9186	0.0365
-6.750	-0.2649	0.01003	0.00471	-0.1059	0.9125	0.0375
-6.500	-0.2377	0.00977	0.00441	-0.1057	0.9053	0.0384
-6.000	-0.1825	0.00935	0.00389	-0.1054	0.8910	0.0404
-5.750	-0.1549	0.00912	0.00360	-0.1052	0.8835	0.0414
-5.500	-0.1275	0.00880	0.00325	-0.1051	0.8751	0.0435
-5.000	-0.0718	0.00845	0.00283	-0.1049	0.8578	0.0476
-4.750	-0.0441	0.00824	0.00259	-0.1047	0.8488	0.0519
-4.500	-0.0162	0.00810	0.00243	-0.1046	0.8388	0.0569
-4.250	0.0117	0.00793	0.00228	-0.1045	0.8288	0.0655
-4.000	0.0394	0.00780	0.00213	-0.1044	0.8184	0.0745
-3.750	0.0674	0.00769	0.00201	-0.1044	0.8073	0.0820
-3.500	0.0954	0.00761	0.00191	-0.1043	0.7964	0.0890
-3.250	0.1232	0.00752	0.00180	-0.1042	0.7851	0.0977
-3.000	0.1512	0.00745	0.00171	-0.1041	0.7733	0.1066
-2.750	0.1791	0.00737	0.00163	-0.1040	0.7616	0.1182
-2.500	0.2069	0.00729	0.00156	-0.1040	0.7497	0.1332
-2.250	0.2346	0.00723	0.00150	-0.1039	0.7378	0.1502
-2.000	0.2625	0.00715	0.00145	-0.1038	0.7254	0.1697
-1.750	0.2903	0.00709	0.00142	-0.1038	0.7132	0.1927
-1.500	0.3180	0.00703	0.00141	-0.1037	0.7012	0.2214
-1.250	0.3456	0.00701	0.00139	-0.1036	0.6886	0.2466
-1.000	0.3734	0.00697	0.00138	-0.1035	0.6754	0.2686
-0.750	0.4012	0.00694	0.00137	-0.1035	0.6626	0.2903
-0.500	0.4288	0.00691	0.00138	-0.1034	0.6497	0.3203
-0.250	0.4562	0.00686	0.00139	-0.1033	0.6365	0.3629
0.000	0.4833	0.00678	0.00141	-0.1032	0.6232	0.4192
0.250	0.5102	0.00658	0.00146	-0.1031	0.6101	0.5177
0.500	0.5366	0.00635	0.00153	-0.1029	0.5975	0.6393
0.750	0.5622	0.00617	0.00160	-0.1024	0.5856	0.7449
1.000	0.5842	0.00594	0.00170	-0.1009	0.5740	0.8717
1.250	0.6163	0.00588	0.00177	-0.1014	0.5622	0.9842
1.500	0.6525	0.00598	0.00181	-0.1033	0.5505	1.0000
1.750	0.6788	0.00611	0.00186	-0.1029	0.5398	1.0000
2.000	0.7055	0.00622	0.00192	-0.1026	0.5294	1.0000
2.250	0.7325	0.00633	0.00199	-0.1024	0.5204	1.0000
2.500	0.7592	0.00646	0.00206	-0.1022	0.5112	1.0000
2.750	0.7865	0.00656	0.00213	-0.1020	0.5029	1.0000
3.250	0.8405	0.00681	0.00231	-0.1016	0.4847	1.0000
3.500	0.8672	0.00696	0.00240	-0.1014	0.4746	1.0000
3.750	0.8941	0.00709	0.00250	-0.1012	0.4646	1.0000
4.000	0.9210	0.00722	0.00260	-0.1010	0.4540	1.0000
4.250	0.9473	0.00739	0.00272	-0.1007	0.4426	1.0000

4.500	0.9734	0.00758	0.00284	-0.1004	0.4273	1.0000
4.750	0.9993	0.00778	0.00297	-0.1001	0.4110	1.0000
5.000	1.0254	0.00797	0.00311	-0.0998	0.3979	1.0000
5.250	1.0518	0.00813	0.00326	-0.0995	0.3861	1.0000
5.500	1.0777	0.00834	0.00342	-0.0992	0.3731	1.0000
5.750	1.1031	0.00857	0.00359	-0.0988	0.3575	1.0000
6.000	1.1280	0.00884	0.00379	-0.0983	0.3398	1.0000
6.250	1.1523	0.00914	0.00401	-0.0978	0.3207	1.0000
6.500	1.1761	0.00948	0.00426	-0.0971	0.2993	1.0000
6.750	1.1988	0.00989	0.00455	-0.0963	0.2737	1.0000
7.000	1.2208	0.01036	0.00488	-0.0954	0.2461	1.0000
7.250	1.2417	0.01089	0.00526	-0.0943	0.2173	1.0000
7.500	1.2614	0.01149	0.00569	-0.0931	0.1865	1.0000
7.750	1.2793	0.01220	0.00621	-0.0915	0.1526	1.0000
8.000	1.2973	0.01288	0.00672	-0.0900	0.1252	1.0000
8.250	1.3164	0.01345	0.00719	-0.0887	0.1065	1.0000
8.500	1.3346	0.01404	0.00769	-0.0872	0.0893	1.0000
8.750	1.3514	0.01469	0.00823	-0.0854	0.0729	1.0000
9.000	1.3676	0.01527	0.00875	-0.0836	0.0622	1.0000
9.250	1.3835	0.01581	0.00926	-0.0817	0.0563	1.0000
9.500	1.4004	0.01631	0.00976	-0.0799	0.0521	1.0000
9.750	1.4171	0.01682	0.01028	-0.0782	0.0491	1.0000
10.000	1.4317	0.01746	0.01091	-0.0762	0.0459	1.0000
10.250	1.4484	0.01797	0.01147	-0.0746	0.0442	1.0000
10.500	1.4653	0.01849	0.01203	-0.0731	0.0427	1.0000
10.750	1.4805	0.01911	0.01267	-0.0714	0.0411	1.0000
11.000	1.4938	0.01986	0.01343	-0.0695	0.0392	1.0000
11.250	1.5061	0.02069	0.01430	-0.0676	0.0376	1.0000
11.500	1.5221	0.02129	0.01495	-0.0662	0.0368	1.0000
11.750	1.5369	0.02199	0.01570	-0.0647	0.0356	1.0000
12.000	1.5500	0.02282	0.01656	-0.0631	0.0343	1.0000
12.250	1.5608	0.02382	0.01758	-0.0614	0.0330	1.0000
12.500	1.5688	0.02506	0.01888	-0.0594	0.0316	1.0000
12.750	1.5831	0.02588	0.01975	-0.0582	0.0308	1.0000
13.000	1.5959	0.02683	0.02075	-0.0569	0.0298	1.0000
13.250	1.6066	0.02796	0.02192	-0.0555	0.0286	1.0000
13.500	1.6141	0.02939	0.02338	-0.0540	0.0273	1.0000
13.750	1.6213	0.03089	0.02494	-0.0526	0.0262	1.0000
14.000	1.6325	0.03209	0.02620	-0.0516	0.0252	1.0000
14.250	1.6414	0.03354	0.02770	-0.0505	0.0241	1.0000
14.500	1.6474	0.03528	0.02947	-0.0493	0.0229	1.0000
14.750	1.6508	0.03731	0.03156	-0.0482	0.0218	1.0000
15.000	1.6585	0.03899	0.03332	-0.0474	0.0209	1.0000
15.250	1.6638	0.04096	0.03533	-0.0466	0.0198	1.0000
15.500	1.6661	0.04330	0.03772	-0.0458	0.0187	1.0000
15.750	1.6666	0.04589	0.04037	-0.0451	0.0179	1.0000
16.000	1.6698	0.04827	0.04284	-0.0447	0.0171	1.0000

16.250	1.6706	0.05099	0.04562	-0.0443	0.0164	1.0000
16.500	1.6692	0.05402	0.04871	-0.0440	0.0157	1.0000
16.750	1.6638	0.05759	0.05235	-0.0439	0.0151	1.0000
17.000	1.6605	0.06101	0.05587	-0.0439	0.0146	1.0000
17.250	1.6584	0.06435	0.05931	-0.0441	0.0142	1.0000
17.500	1.6548	0.06793	0.06298	-0.0444	0.0138	1.0000
17.750	1.6497	0.07175	0.06689	-0.0448	0.0134	1.0000
18.000	1.6430	0.07583	0.07106	-0.0453	0.0131	1.0000
18.250	1.6346	0.08024	0.07555	-0.0461	0.0128	1.0000
18.500	1.6237	0.08507	0.08047	-0.0470	0.0124	1.0000
18.750	1.6097	0.09040	0.08590	-0.0482	0.0121	1.0000

8.2 Code

For the sake of brevity, only the code files that are key to the analysis are included below. However, in the spirit of completeness, the repository containing the complete data, source code, and notes for this report can be found at [github:vaisriv/ENAE464-lab02](https://github.com/vaisriv/ENAE464-lab02).

Listing 8.2.1: Index File

```
./src/index.py
```

```

1 import os
2 import sys
3 import csv
4 import numpy as np
5 import matplotlib.pyplot as plt
6
7 def main():
8     # — Physical constants & ambient conditions ——————
9     P_AMB      = 100900.0 # Ambient pressure [Pa]
10    T_AMB      = 298.15   # Ambient temperature [K]
11    RHO_AIR    = 1.18     # Air density [kg/m^3]
12    RHO_WATER  = 998.0    # Water density [kg/m^3]
13    G          = 9.81     # Gravitational acceleration [m/s^2]
14
15    # — File paths ——————
16    INPUT_CSV  = "./data/pressure_vs_theta.csv"
17    OUTPUT_CSV = "./outputs/text/pressure_vs_theta.csv"
18    OUTPUT_FILE = "./outputs/text/summary.txt"
19
20    # — Ensure figures output directory exists ——————
21    FIG_DIR = os.path.join("outputs", "figures")
22    os.makedirs(FIG_DIR, exist_ok=True)
23
24    # — Read CSV ——————
25    theta_deg_list = []
26    P_inf_raw_list = []
27    P_0_raw_list   = []

```

```

28 P_raw_list      = []
29
30 with open(INPUT_CSV, newline='') as f:
31     reader = csv.DictReader(f)
32     for row in reader:
33         theta_deg_list.append(float(row["theta_deg"]))
34         P_inf_raw_list.append(float(row["P_inf"]))
35         P_0_raw_list.append(float(row["P_0"]))
36         P_raw_list.append(float(row["P"]))
37
38 N = len(theta_deg_list)
39 print(f"Read {N} data points from '{INPUT_CSV}'")
40
41 # — Convert water column heights to differential pressures ——————
42 # The manometer readings are heights (in cm of water):
43 #   ΔP = ρ_water · g · Δh
44 #   P_actual = P_ref - ρ_water · g · h_reading
45 #   P_surface - P_freestream = ρ_water · g · (h_surface - h_inf)
46 #   q_inf = P_θ - P_inf = ρ_water · g · (h_θ - h_inf)
47 SCALE = 1e-2 # Measurements taken in cm (convert to meters)
48
49 # — Compute differential pressures ——————
50 # For each data point we have a corresponding P_inf and P_θ reading,
51 # so we compute per-row to account for any drift.
52 delta_P_list = [] # P_surface - P_freestream [Pa]
53 q_inf_list   = [] # dynamic pressure [Pa]
54 U_inf_list   = [] # freestream velocity [m/s]
55 Cp_list       = [] # pressure coefficient
56
57 for i in range(N):
58     h_inf = P_inf_raw_list[i] * SCALE # freestream static tap height [m]
59     h_θ   = P_0_raw_list[i]   * SCALE # stagnation tap height [m]
60     h_p   = P_raw_list[i]     * SCALE # surface tap height [m]
61
62     # (P_surface - P_inf) in Pa
63     # Higher reading = higher pressure, so:
64     delta_P = RHO_WATER * G * (h_p - h_inf)
65
66     # Dynamic pressure q = P_total - P_static = rho_w * g * (h_θ - h_inf)
67     q_inf = RHO_WATER * G * (h_θ - h_inf)
68
69     if q_inf ≤ 0:
70         print(f"WARNING row {i}: q_inf = {q_inf:.2f} Pa ≤ 0 (h_inf={h_inf:.4f},
71             ↪ h_θ={h_θ:.4f})")
72         # Use absolute value as fallback but flag it
73         q_inf = abs(q_inf) if abs(q_inf) > 1e-6 else 1e-6
74
75     U_inf = (2.0 * q_inf / RHO_AIR) ** 0.5 # Bernoulli: q = 0.5 * rho * U^2

```

```

75
76     Cp = delta_P / q_inf #  $C_p = (P - P_{\text{inf}}) / q_{\text{inf}} = (P - P_{\text{inf}}) / (0.5 * \rho * U^2)$ 
77
78     delta_P_list.append(delta_P)
79     q_inf_list.append(q_inf)
80     U_inf_list.append(U_inf)
81     Cp_list.append(Cp)
82
83     # — Convert lists to numpy arrays for convenience ——————
84     theta_deg = np.array(theta_deg_list)
85     theta_rad = np.deg2rad(theta_deg)
86     Cp_exp = np.array(Cp_list)
87     dP_exp = np.array(delta_P_list)
88     q_inf_arr = np.array(q_inf_list)
89
90     # — Inviscid (potential flow) theory for a cylinder ——————
91     #  $C_{p,\text{inviscid}} = 1 - 4 \sin^2(\theta)$ 
92     theta_theory = np.linspace(0, 180, 500)
93     theta_theory_rad = np.deg2rad(theta_theory)
94     Cp_inviscid = 1.0 - 4.0 * np.sin(theta_theory_rad) ** 2
95
96     # — Figure 1:  $C_p$  vs  $\theta$  ——————
97     fig1, ax1 = plt.subplots(figsize=(9, 5))
98     ax1.plot(theta_deg, Cp_exp, 'o', markersize=5, label='Experimental')
99     ax1.plot(theta_theory, Cp_inviscid, '-', linewidth=1.5, label='Inviscid theory ($1 - 4 \sin^2(\theta)$)')
100    ax1.set_xlabel(r'$\theta$ [deg]')
101    ax1.set_ylabel(r'$C_p$')
102    ax1.set_title(r'Pressure Coefficient $C_p$ vs Angular Position $\theta$')
103    ax1.legend()
104    ax1.grid(True, alpha=0.3)
105    fig1.tight_layout()
106    fig1.savefig(os.path.join(FILE_DIR, "Cp_vs_theta.png"), dpi=300)
107    print(f"Saved {os.path.join(FILE_DIR, 'Cp_vs_theta.png')}")

108
109     # — Figure 2:  $C_d$  vs  $\theta$  (cumulative drag coefficient) ——————
110     #
111     # The drag force on the cylinder (per unit span) is:
112     #  $D = R \int_0^{2\pi} P \cdot \cos(\theta) d\theta$ 
113     #
114     # We define the drag coefficient as:
115     #  $C_d = D / (q_{\text{inf}} \cdot d) = D / (q_{\text{inf}} \cdot 2R)$ 
116     #
117     # Substituting and non-dimensionalising with  $C_p = (P - P_{\text{inf}}) / q_{\text{inf}}$  :
118     #  $C_d = (1/2) \int_0^{2\pi} C_p \cdot \cos(\theta) d\theta$ 
119     #
120     # (The  $P_{\text{inf}}$  contribution integrates to zero over a closed surface.)
121     #

```

```

122 # We approximate the integral cumulatively using the trapezoidal rule so we
123 # can plot C_d as a function of the upper integration limit θ.
124
125 # Sort by angle to ensure proper integration order
126 sort_idx = np.argsort(theta_deg)
127 theta_sorted = theta_deg[sort_idx]
128 theta_sorted_rad = theta_rad[sort_idx]
129 Cp_sorted = Cp_exp[sort_idx]
130
131 # Integrand: Cp(θ) · cos(θ)
132 integrand_exp = Cp_sorted * np.cos(theta_sorted_rad)
133
134 # Cumulative trapezoidal integration: (1/2) ∫θ^θ' Cp·cos(θ') dθ'
135 Cd_cumulative_exp = np.zeros(len(theta_sorted))
136 for j in range(1, len(theta_sorted)):
137     dtheta = theta_sorted_rad[j] - theta_sorted_rad[j - 1]
138     Cd_cumulative_exp[j] = Cd_cumulative_exp[j - 1] + 0.5 * (integrand_exp[j] +
139         ↳ integrand_exp[j - 1]) * dtheta
140 Cd_cumulative_exp *= 0.5 # the 1/2 prefactor from C_d = (1/2) ∫ Cp cos(θ) dθ
141
142 # Inviscid theory: Cp_inv = 1 - 4sin²θ → Cd = θ (d'Alembert's paradox)
143 # Cumulative for plotting:
144 integrand_inv = (1.0 - 4.0 * np.sin(theta_theory_rad) ** 2) * np.cos(theta_theory_rad)
145 Cd_cumulative_inv = np.zeros(len(theta_theory))
146 for j in range(1, len(theta_theory)):
147     dtheta = theta_theory_rad[j] - theta_theory_rad[j - 1]
148     Cd_cumulative_inv[j] = Cd_cumulative_inv[j - 1] + 0.5 * (integrand_inv[j] +
149         ↳ integrand_inv[j - 1]) * dtheta
150 Cd_cumulative_inv *= 0.5
151
152 fig2, ax2 = plt.subplots(figsize=(9, 5))
153 ax2.plot(theta_sorted, Cd_cumulative_exp, 'o-', markersize=4, linewidth=1,
154     ↳ label='Experimental (trapezoidal)')
155 ax2.plot(theta_theory, Cd_cumulative_inv, '-', linewidth=1.5, label="Inviscid theory
156     ↳ (d'Alembert: $C_d = 0$)")
157 ax2.set_xlabel(r'$\theta$ [deg]')
158 ax2.set_ylabel(r'$C_d(\theta)$ (cumulative)')
159 ax2.set_title(r'Cumulative Drag Coefficient $C_d$ vs Angular Position $\theta$')
160 ax2.legend()
161 ax2.grid(True, alpha=0.3)
162 fig2.tight_layout()
163 fig2.savefig(os.path.join(FIG_DIR, "Cd_vs_theta.png"), dpi=300)
164 print(f"Saved {os.path.join(FIG_DIR, 'Cd_vs_theta.png')}")

# Print final Cd value (full integral over θ to max angle)
print(f"\nExperimental C_d (integrated θ to {theta_sorted[-1]:.0f} deg):
165     ↳ {Cd_cumulative_exp[-1]:.4f}")

```

```
164     print(f"Inviscid theory C_d (integrated 0 to 360 deg): {Cd_cumulative_inv[-1]:.6f}\n"
165           "≈ 0, d'Alembert's paradox)")
166
167 # — Summary statistics ——————
168 q_avg    = sum(q_inf_list) / N
169 U_avg    = sum(U_inf_list) / N
170
171 with open(OUTPUT_FILE, "w", newline="") as f:
172     f.write(f"Ambient conditions:")
173     f.write(f" P_amb    = {P_AMB} Pa")
174     f.write(f" T_amb    = {T_AMB} K")
175     f.write(f" rho_air = {RHO_AIR} kg/m^3")
176     f.write(f"\nDerived quantities (averages over all rows):")
177     f.write(f" q_inf    = {q_avg:.2f} Pa")
178     f.write(f" U_inf    = {U_avg:.2f} m/s")
179
180 print(f"\nSummary written to '{OUTPUT_FILE}'.")
181
182 # — Write output table ——————
183 with open(OUTPUT_CSV, "w", newline="") as f:
184     writer = csv.writer(f)
185     writer.writerow(["theta_deg", "P-P_inf_Pa", "Cp", "P_Pa"])
186
187     for i in range(N):
188         # P_actual = P_amb + (P_surface - P_inf) i.e. ambient + differential
189         P_actual = P_AMB + delta_P_list[i]
190         writer.writerow([
191             f"{theta_deg_list[i]:.1f}",
192             f"{delta_P_list[i]:.3f}",
193             f"{Cp_list[i]:.4f}",
194             f"{P_actual:.3f}",
195         ])
196
197 print(f"\nResults written to '{OUTPUT_CSV}'.")
198
199 if __name__ == "__main__":
200     main()
```