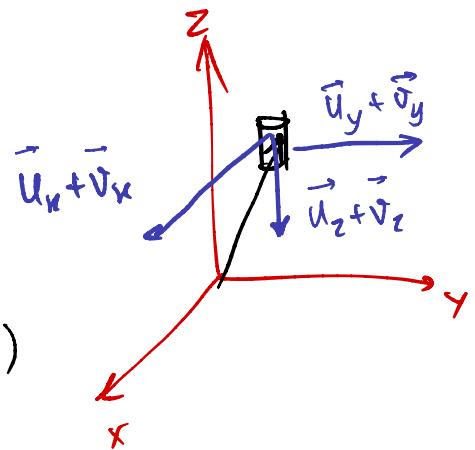


CUBE PARACHUTE EMPERICAL FORMULA

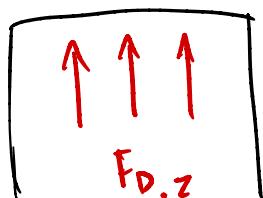
Set u_x, u_y, u_z = natural wind

Set $v_z(t)$ = descent rate

Set $v_{x(t)}, v_{y(t)}$ = difference of rate (lateral)



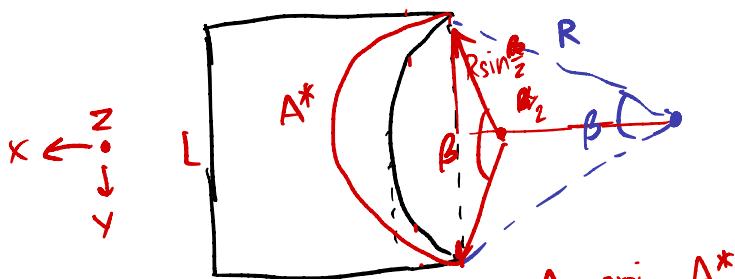
CASE 1: $\|U_z\|, \|U_y\|, \|U_x\| = 0$



$$F_{D,z} = \frac{1}{2} \rho C_D A \cdot v^2 ; \quad C_D = 0.54 \quad A = \frac{\pi}{3} L^2$$

$$F_{D,z} = 0.63 \rho L^2 v^2 \quad \rho \propto R^{-1}$$

CASE 2: $\|U_x\| > 0, \|U_y\|, \|U_z\| = 0$



$$2R \sin \frac{\alpha}{2} = \frac{L}{2} \rightarrow \alpha = 2 \sin^{-1} \left(\frac{L}{2R} \right) \quad R = \frac{k}{u_x} \quad k = \theta$$

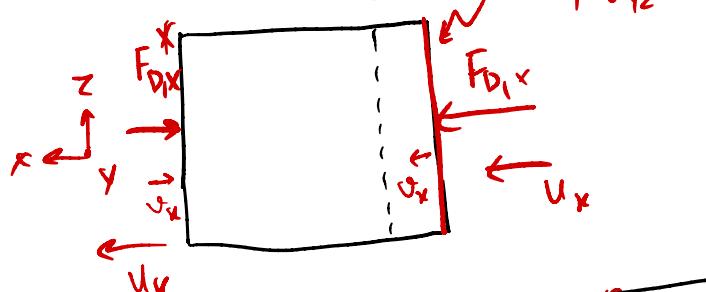
$$A_z = L^2 - \left(\frac{1}{2} \alpha R^2 - R \sin \frac{\alpha}{2} \cos \frac{\alpha}{2} \right)$$

$$A_z = L^2 - \frac{R^2}{2} (\alpha - \sin \alpha)$$

$$F_{D,x} = 0.63 \rho L^2 v_x^2 \Delta A_z$$

$$F_{D,z} = 0.63 \rho \left(L^2 - \frac{R^2}{2} (\alpha - \sin \alpha) \right) v_z^2$$

if $k = \theta$ (shape parameter)



$$F_{D,z} = 0.63 \rho \left(L^2 - \frac{\theta^2}{2 u_x} \left(2 \sin^{-1} \left(\frac{u_x L}{2} \right) - u_x L \sqrt{1 - \left(\frac{u_x L}{2} \right)^2} \right) \right) v_z^2 \quad (**)$$

$$F_{D,x} = \underbrace{0.63 \rho L^2 v_x^2}_{\alpha} \quad (*)$$

air motion air (*)

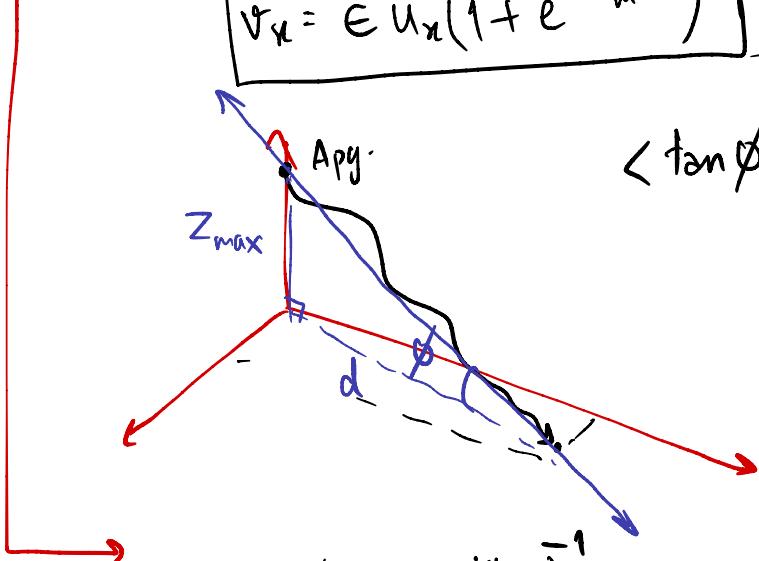
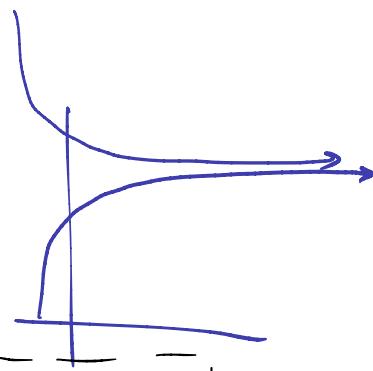
$$x: m \frac{dv_x}{dt} = F_{D,x} - F_{D,x}^* \\ = \alpha(u_x - v_x)^2 - \alpha(u_x + v_x)^2 \\ = -\alpha(2v_x)(2u_x)$$

$$m \frac{dv_x}{dt} = -4\alpha u_x v_x$$

$$\frac{1}{v_x} dv_x = -\frac{4\alpha u_x}{m} dt$$

$$\ln \left| \frac{v_x}{v_0} \right| = -\frac{4\alpha u_x}{m} t + C_0$$

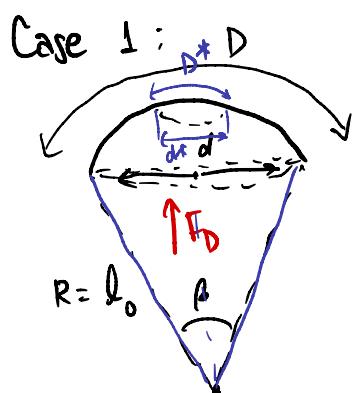
$$\therefore \begin{cases} v_x = E u_x \left(1 - e^{-\frac{4\alpha u_x}{m} t} \right) & ; \quad v_x < u_x \\ v_x = E u_x \left(1 + e^{-\frac{4\alpha u_x}{m} t} \right) & ; \quad v_x > u_x \end{cases}$$



$$\langle \tan \phi \rangle_{set} = \frac{z_{max}}{v_n t}$$

$$E = \frac{v_x}{u_x} \left(1 \pm e^{-\frac{4\alpha u_x}{m} t} \right)^{-1} \leftrightarrow E = \frac{v_x}{u_x} \gamma^{-1}$$

TEST FOR CIRCULAR VENTED PARACHUTE



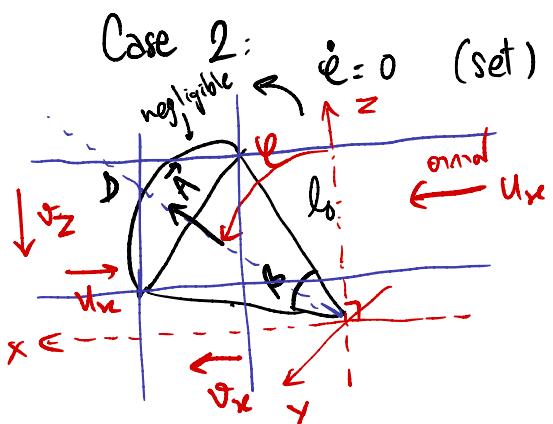
$$l_0 \beta = D \rightarrow \beta = \frac{D}{l_0}$$

$$A_{\text{proj}} = \pi \left(\frac{d}{2}\right)^2; \quad \frac{d}{2} = l_0 \sin\left(\frac{\beta}{2}\right)$$

$$= \pi l_0^2 \sin^2\left(\frac{\beta}{2}\right) \quad \approx \frac{\pi D^2}{4}; \quad l_0 \rightarrow \infty$$

$$A_{\text{proj}} = \pi l_0^2 \sin^2\left(\frac{D}{2l_0}\right) \quad * \quad l_0 > \frac{D}{2}$$

$$F_{D,z} = \frac{1}{2} \rho C_D \pi l_0^2 \sin^2\left(\frac{D}{2l_0}\right) v^2$$



$$\text{then } F_{D,z} \text{ on } A_{\text{proj}} \approx A_{\text{proj}} \cos \theta$$

(mean proj_{xy}(Vec A))

$$\text{then } F_{D,z} = \alpha v_z^2 \cos \theta, \quad F_{D,x} = \alpha v_z^2 \sin \theta$$

\$\Rightarrow\$ tan \$\theta\$ = \$\frac{\sum v_z}{\sum v_x}\$

$$\text{then } \tan \theta = \infty \frac{\sum v_z}{\sum v_x}$$

$$\therefore \cos \theta = \frac{u_x + v_x}{\sqrt{v_z^2 + (u_x + v_x)^2}} \rightarrow \text{then } F_{D,z} \text{ is given} \quad \tan \theta = \frac{v_z}{u_x + v_x}$$

$$\sin \theta = \frac{v_z}{\sqrt{v_z^2 + (u_x + v_x)^2}}$$

$$\text{let } v_x: \quad m \frac{dv_x}{dt} = F_{D,x} - F_{D,z}^*$$

$$= \alpha \cos \theta [(u_x - v_x)^2 - (u_x + v_x)^2]$$

* For vented paragliders:

$$\text{use } A_{\text{proj}} = \pi l_0^2 \left(\sin^2\left(\frac{D}{2l_0}\right) - \sin^2\left(\frac{d}{2l_0}\right) \right)$$

if D is OD, d is vent D.

Sol'n:

$$\boxed{v_x' + \frac{4\alpha u_x}{m} v_x \left(\frac{u_x + v_x}{\sqrt{v_z^2 + (u_x + v_x)^2}} \right) = 0}$$

Approximation (special case : $\theta = 45^\circ$; $\cos \theta = \frac{1}{\sqrt{2}}$)

$$\left\{ \begin{array}{l} v_x' + \frac{4du_x}{m} v_x \left(\frac{1}{\sqrt{2}} \right) = 0 : v_x = u_x \left(1 \pm \bar{e}^{-\frac{2\sqrt{2}du_x}{m} t} \right) \end{array} \right.$$

$$EV_z = u_x + v_x \rightarrow v_x = EV_z - u_x$$

$$E = \frac{u_x}{v_z} \left(2 \pm e^{-\frac{2\sqrt{2}du_x}{m} t} \right) \quad \text{time factor}$$

$$E = \frac{u_x}{v_z} (1 + \gamma)$$